

## Running SU<sup>2</sup>

SU<sup>2</sup> Release Version 2.0 Workshop Stanford University Tuesday, January 15<sup>th</sup>, 2013

Sean Copeland & Tom Taylor

Department of Aeronautics & Astronautics Stanford University





#### **Outline**

- SU<sup>2</sup> environment variable
- What do I need to run a simulation?
- Test case definition
- Interactive session
- Questions?





#### SU<sup>2</sup> Environment Variables



- Did you set your SU<sup>2</sup> environment variables during the installation procedure?
- i.e., in .bashrc:

```
# Set the SU2 source code home directory
export SU2_HOME=/Users/tomtaylor/SU2

# Identify directory where SU2 executables and python scripts are stored
Export SU2_RUN=/Users/tomtaylor/SU2Py

# Add SU2_RUN to the $PATH
Export PATH=$PATH:$SU2_RUN
```



### Running Simulations with SU<sup>2</sup>



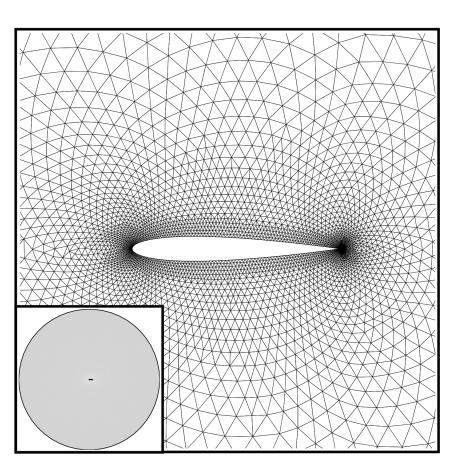
- What do I need to run simulations with SU<sup>2</sup>?
  - Configuration file (.cfg)
  - Mesh file (.su2 or .cgns)
- This session will use:
  - inv\_NACA0012.cfg
  - Mesh\_NACA0012\_inv.su2
- These are found in
  - > User Guide > Quick Start Tutorial
    /SU2/TestCases/inv\_NACA0012/



# Test Case NACA 0012 Airfoil



- Transonic, Euler flow
- Mach No. = 0.8
- Pressure =  $101,325 \text{ Nm}^{-2}$
- Temperature = 273.15K
- Angle of attack = 1.25°





#### Flow Solution



#### Config options:

```
PHYSICAL_PROBLEM= EULER
%
MATH_PROBLEM= DIRECT
%
MACH_NUMBER= 0.8
%
AoA= 1.25
%
FREESTREAM_PRESSURE= 101325.0
%
FREESTREAM_TEMPERATURE= 273.15
%
MESH_FILENAME= mesh_NACA0012_inv.su2
```

- Most parameters have default values
- Order of config options is not important



#### Restart



- Simulations can be restarted from partially converged results
- Config options:

```
RESTART_SOL= NO %
EXT_ITER= 50
```

```
RESTART_SOL= YES %
SOLUTION_FLOW_FILENAME= solution_flow.dat
```



#### Solver Parameters



- Among the many options in the config file, various parameters exist to modify the solution method
- Config options:

```
RESTART_SOL= NO %
CFL_NUMBER= 4.0 %
CFL_RAMP= ( 1.1, 10, 10.0 ) %
CONV_NUM_METHOD_FLOW= ROE-2ND_ORDER
```



### **Adjoint Solution**



- Sensitivity of a functional to changes in the flow
  - e.g., How does changing the airfoil shape affect lift?
- Additional required file:
  - Converged flow solution
- Config options:

```
MATH_PROBLEM= ADJOINT

%

RESTART_SOL= NO

%

ADJOINT_TYPE= CONTINUOUS

%

ADJ_OBJFUNC= DRAG

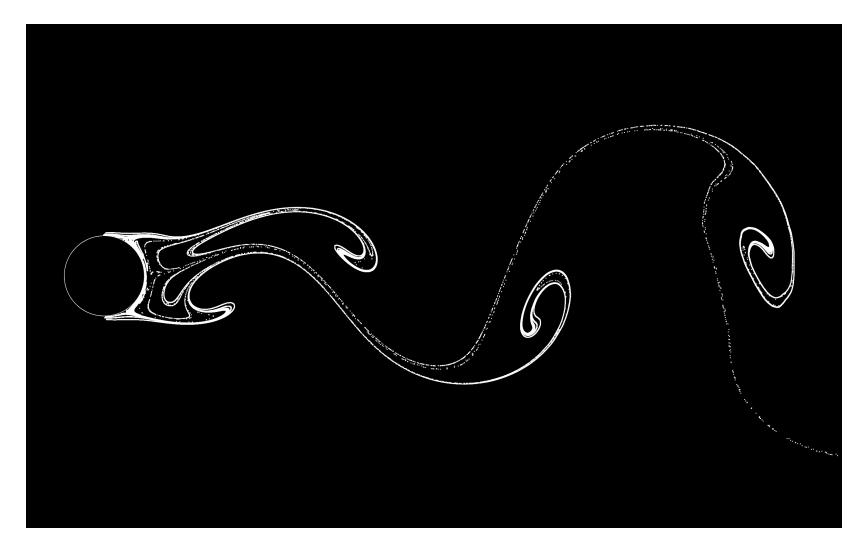
%

SOLUTION_FLOW_FILENAME= solution_flow.dat
```



# Example: Separated Flow Over Cylinder

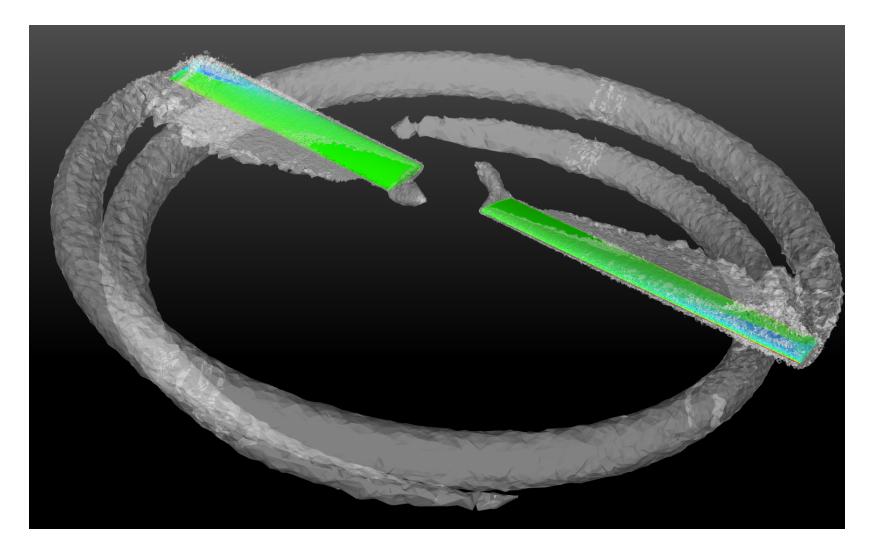






# Example: Tip Vortices From Rotor







#### **Additional Resources**



- Online documentation
  - http://su2.stanford.edu
- Online tutorials
  - > User Guide > User's Guide > User's Tutorials
- TestCases directory
  /SU2/TestCases/
- CFD Online forum

http://www.cfd-online.com/Forums/su2/