

Astrometric Orbit Determination

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Abstract

Astrometry is used to measure properties of celestial objects such as their position on the sky, their movement, velocities and acceleration. In this experiment, we used a charged coupled device called CCD that mapped the sky on a 2D plane. Observations were made using the Dunlap Institute Telescope located in New Mexico, United States. The object of interest for this experiment is an asteroid named 30 Urania. We measured the plate constants of the CCD in order to correct for the non-ideal CCD. Then we applied these plate constants to the data from 30 Urania over the course of four nights in January 2012 and measured the corresponding right ascension and declination of this asteroid in order to determine the orbit of motion of the asteroid and to find its orbital elements such as the inclination, its true, mean and eccentric anomaly, its semi major axis and .

1 Introduction

Nowadays, with the advancements of technology, we are able to use computers to do astrometry in order to obtain position and motion of celestial objects. We have used the stellar map of NGC7331 taken with DIT and found the centroid points and compared and matched them with USNO-B1.0 catalogue in order to find the plate constants of the CCD using least square fitting technique. Here in this lab, we will be using the celestial coordinates(α and

δ) obtained by the conversion from plate constants which is calculated separately for each night(for more precision) in order to find the orbital elements of the asteroid 30 Urania, calculate its distance relative to the Earth and the Sun and to determine its orbit for future time.

2 Observation and Data

2.1 Equipment

We used the Dunlap Institute Telescope(DIT) which is a 50-cm robotic telescope dedicated to search for optical counterparts of gamma-ray bursts. The telescope was located at New Mexico on Mt. Joy(32:54:10 N +105:31:46 W). The telescope is equipped with a 4096×4096 pixel CCD array and the resultant field of view is approximately 36×36 arc minutes. Pixel size of this CCD is 9μ . This CCD operated in binning mode to collect the data we will be using through this experiment. Binning is useful since it reduces the size of the resultant FITS file and reduces the effect of read-out noise. For 2×2 binning the resultant array is 2048×2048 pixels.

2.2 CCD Calibration

In order to have good images from the CCD we need to calibrate them. A good image is one that is dark subtracted and divided by flat frames. Therefore, in order to use the data from the CCD, we subtracted the dark data from the actual data. Dark data was taken with the same exposure time as the actual data while the shutter was closed. Dark current is produced due to the noise in the CCD and it might come from the thermal energy of the electrons; therefore, CCDs are kept in cool places in order to reduce the noise. We also divided the dark subtracted data by the flat frames. Flats are morning and evening twilight sky images that are used to normalize the image. Another factor to consider are the bias frames, which are zero second exposures that provide a measure of digital offset when the signal is zero. Here, the bias is assumed to be zero for this long exposure with DIT.

3 Calculation and Modeling

3.1 Asteroid Position

The coordinates of astronomical objects are usually measured by the angles right ascension, α and declination, δ , in the equatorial system defined by reference stars measured in Interna-

tional Celestial Reference System, taken at epoch 2000. Therefore, the cartesian coordinates are give by equation (...) below.

$$x_{eq} = \cos \alpha \cos \delta, y_{eq} = \sin \alpha \cos \delta, z_{eq} = \sin \delta \quad (1)$$

We can convert the equatorial coordinate to ecliptic coordinate by a transformation along x-axis by the angle ϵ by equation(.....) given below.

$$s = \begin{pmatrix} x \\ y \\ z \end{pmatrix} = \begin{pmatrix} 1 & 0 & 0 \\ 0 & \cos \epsilon & \sin \epsilon \\ 0 & -\sin \epsilon & \cos \epsilon \end{pmatrix} \begin{pmatrix} x_{eq} \\ y_{eq} \\ z_{eq} \end{pmatrix} \quad (2)$$

where $\epsilon = 23^\circ.43929111$ is the tilt of the Earth for the equinox 2000. s is in the ecliptic Cartesian frame. In this manner, we are able to calculate the s_1 , s_2 and s_3 vectors and thereafter, we can calculate the first and second derivatives of s_2 vector by equation (...) given below:

$$\dot{s}_2 = \frac{\tau_3(s_2 - s_1)}{\tau_1(\tau_1 + \tau_3)} + \frac{\tau_1(s_3 - s_2)}{\tau_1(\tau_1 + \tau_3)} \quad (3)$$

$$\ddot{s}_2 = \frac{2(s_3 - s_2)}{\tau_3(\tau_1 + \tau_3)} + \frac{2(s_2 - s_1)}{\tau_1(\tau_1 + \tau_3)} \quad (4)$$

The next step is to find the \mathbf{R} vector. We obtain the components of the \mathbf{R} vector from the JPL Horizons Ephemeris web interface for the desired date which is 24th, 25th and 26th of January 2012. The \mathbf{R} vector components are given in the table (...).

Table 1: Date and position of asteroid 30 Urania in the sky

UT Date 2012/01	Julian Date	Right Ascension(radians)	Declination(radians)
19	2455946.68646	0.77589496	0.33553406
20	2455947.69476	0.77991461	0.33640024
22	2455949.73866	0.78825718	0.33765579
23	2455950.68528	0.79249126	0.33851884

Table 2: Date and position of asteroid 30 Urania in the sky obtained from JPL

UT Date 2012/01	Julian Date	Right Ascension(radians)	Declination(radians)
19	2455946.68646	0.775884664	0.335833345
20	2455947.69476	0.779874196	0.33648493
22	2455949.73866	0.788401584	0.337913196
23	2455950.68528		

Table 3: Sun-Earth vector Cartesian components of \mathbf{R} in ecliptic coordinates [Au]

X	Y	Z
-0.4820742567460204	0.8578010618775774	$-2.425231096584922 \times 10^{-5}$
-0.4974148410812385	0.8490994851434150	$-2.462584044959891 \times 10^{-5}$
-0.5280130493076013	0.8306420761670466	$-2.515673232526080 \times 10^{-5}$
-0.5419538025371889	0.8217251214639645	$-2.524538626668697 \times 10^{-5}$

We also need to find the magnitude of the \mathbf{R} vector for the second data(since the second night is the one we use to find the position of the asteroid as we need three sets of data for figuring out the position). \mathbf{R} vector magnitude can be calculated by squaring each of the components and taking the square root of their sum.

$$R = \sqrt{R_x^2 + R_y^2 + R_z^2} \quad (5)$$

We now have all the values we need and should be able to calculate ρ value by the equation (...):

$$\rho = k^2 \left(\frac{1}{R^3} - \frac{1}{r^3} \right) \frac{\dot{\mathbf{s}} \cdot (\mathbf{R} \times \mathbf{s})}{\dot{\mathbf{s}} \cdot (\dot{\mathbf{s}} \times \mathbf{s})} \quad (6)$$

We do not have the value of r explicitly yet; however, we can guess an initial r value and write a for loop to calculate ρ until it converges using the new values for r . This part of the code I used can be found in the appendix.(....)

After finding the ρ value, we can calculate the r value by equation (...):

$$r^2 = \rho^2 + R^2 + 2\rho\mathbf{R} \cdot \mathbf{s} \quad (7)$$

Note that this r is different from the one we just guessed in equation (.....). We can also calculate derivative of ρ using equation (....) :

$$\dot{\rho} = \frac{k^2}{2} \left(\frac{1}{R^3} - \frac{1}{r^3} \right) \frac{\ddot{\mathbf{s}} \cdot (\mathbf{R} \times \mathbf{s})}{\dot{\mathbf{s}} \cdot (\dot{\mathbf{s}} \times \mathbf{s})} \quad (8)$$

We can now calculate the position vector of the asteroid with respect to sun using simple vector addition:

$$\mathbf{r} = \mathbf{R} + r\mathbf{s} \quad (9)$$

We may also find the derivative of the \mathbf{R} vector using the same way we found the derivative of the vector \mathbf{s} , which can be written as equation (...):

$$\dot{\mathbf{R}}_2 = \frac{\tau_3(\mathbf{R}_2 - \mathbf{R}_1)}{\tau_1(\tau_1 + \tau_3)} + \frac{\tau_1(\mathbf{R}_3 - \mathbf{R}_2)}{\tau_1(\tau_1 + \tau_3)} \quad (10)$$

In order to calculate the velocity of the asteroid relative to the sun we need to find the derivative of the \mathbf{r} vector we found in equation (...) and employ equation (...):

$$\dot{\mathbf{r}} = \dot{\mathbf{R}} + \rho\dot{\mathbf{s}} + \dot{\rho}\mathbf{s} \quad (11)$$

3.2 Orbital Elements

We can now use the values we got from previous section to find the orbital elements of 30 Urania using Laplace's method. Since we know the magnitude of the radius vector, r , the asteroid velocity, $V = |\dot{\mathbf{r}}|$ and $k=0.017202098950$, we can compute the semimajor axis by equation (...):

$$a = \frac{k^2 r}{2k^2 - rV^2} \quad (12)$$

Then we can compute the \mathbf{h} vector components by the cross product of the \mathbf{r} and $\dot{\mathbf{r}}$ vectors:

$$\mathbf{h} = \mathbf{r} \times \dot{\mathbf{r}} \quad (13)$$

Using the \mathbf{h} vector components, we can compute Ω and i by equations (...) and (...) respectively:

$$\tan \Omega = \frac{-h_x}{h_y} \quad (14)$$

$$\cos i = \frac{h_z}{h} \quad (15)$$

In order to find the eccentricity, we use equation (...) given below:

$$e = \sqrt{1 - \left(\frac{h^2}{ak^2}\right)} \quad (16)$$

In order to compute the last two Keplerian orbital elements-the argument of perihelion, ω , and the epoch of perihelion, τ , we need to calculate the eccentric and true anomalies. Equation (....) and (...) give the eccentric and true anomalies, respectively.

$$\cos E = \frac{a - r}{ae} \quad (17)$$

The sign of E depends on the body's radial velocity; $E \geq 0$ when the radial velocity is positive(between perihelion and aphelion) and $E \leq 0$ when the radial velocity is negative(from aphelion to perihelion).

$$v = 2 \arctan\left(\sqrt{\frac{1+e}{1-e}} \tan E/2\right) \quad (18)$$

$$\cos(v + \omega) = \frac{x \cos \Omega + y \sin \Omega}{r} \quad (19)$$

For the true anomaly, τ we can use equation (....) :

$$M = n(t - \tau) \quad (20)$$

where $n = ka^{-3/2}$ is the mean motion and t is the epoch of the second observation. The calculated orbital elements are listed in table (...) along with their true values and the error in each. Since the error is small, we can be confident about our method and values. However, during this lab we found out that this method is very sensitive to the coordinates of the asteroid obtained via transformation by plate constants. At first, we were using values that were similar to the true values (JPL values) within two decimal places, but the values of \mathbf{r} and ρ were converging to 0 and 1 respectively, which meant that the asteroid is somewhere on the surface of the Earth. After computing more accurate right ascension and declination of the asteroid using more precise plate constants with smaller $\chi^2_{reduced}$, the values of the RA and Dec of the asteroid matched with the JPL values within 4 decimal places and we were able to get good estimate of the position of the asteroid. In order to make sure that our method and equations are correct, we first applied the same method and equations to the asteroid Ceres and compared the orbital elements we got with the ones given in the lab handout and they matched with small residuals; therefore, we were confident about the method and the equations used to compute the orbital elements of 30 Urania.

Table 4: Sun-Earth vector Cartesian components of \mathbf{R} in ecliptic coordinates [Au]

	a	Ω	i	e	ω	τ
	[Au]	[deg]	[deg]		[deg]	[Julian Date]
True	2.766	80.72	10.61	0.179	73.12	2454785.970
Estimated	3.186	80.68	10.50	0.079		2454868
Error	-0.42	0.04	0.11	0.09		82.03

3.3 Orbital Motion of Asteroid Ceres

In this sub section, we will be using Keplerian orbital elements provided in the lab handout to compute and plot some of the orbital elements such as true anomaly and the orbital separation, r , as a function of time. We first need to compute the mean anomaly by equation (...). Then using Taylor series we are able to find the E_n numerically by the following equation:

$$E_{n+1} = E_n + \frac{M - M(E_n)}{1 - e \cos(E_n)} \quad (21)$$

We can then plot the eccentric and true anomalies as a function of time in Julian Date. The result is shown in figure (...).

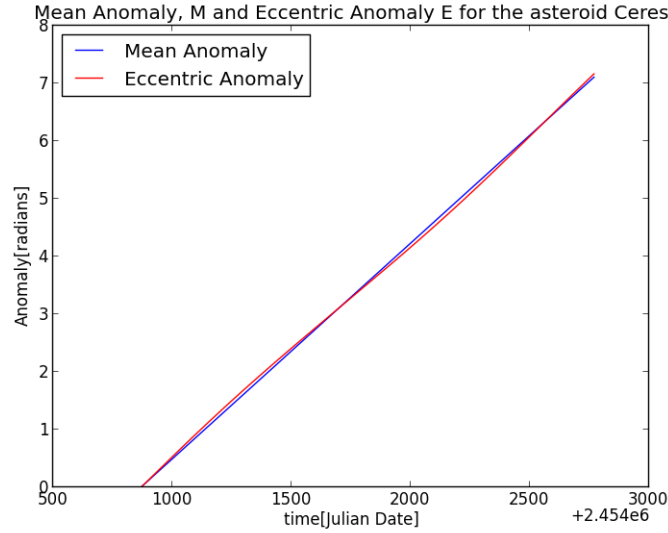


Figure 1: This figure shows the mean anomaly, M , and eccentric anomaly, E , for the asteroid Ceres using the orbital elements tabulated in table (...). The eccentricity of Ceres is small ($e=0.079$) and the difference between these two angles is small.

We can then compute the true anomaly by equation (...) and since we have both v and

w, we are able to calculate the polar angle by:

$$\theta = v + \omega \quad (22)$$

and the Cartesian components of the position of the asteroid are given by equation (...):

$$r = \begin{pmatrix} r \cos \theta \\ r \sin \theta \\ 0 \end{pmatrix} \quad (23)$$

The plot of the r as a function of time is shown in figure (...):

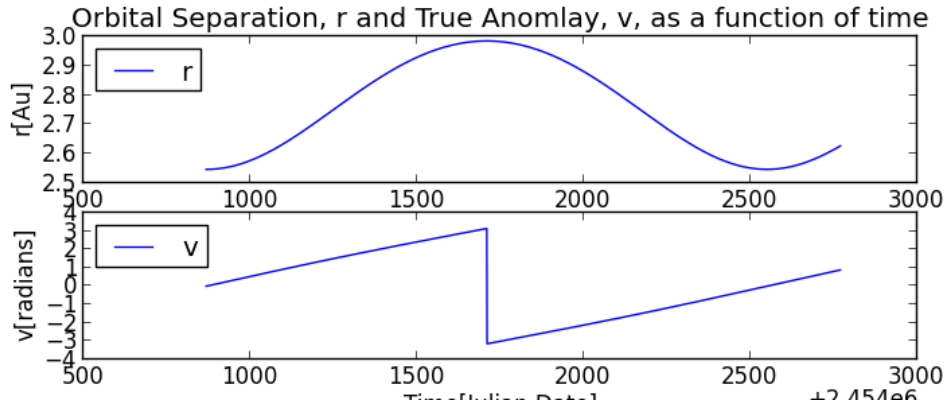


Figure 2: This figure shows the orbital separation and true anomaly of the asteroid Ceres respectively over one period.

3.4 Orbital Motion of Asteroid 30 Urania

The same way we predicted the position of Ceres for future time, we can determine the orbit of motion of 30 Urania using the orbital elements we computed in the previous sections. Table (...) contains the orbital elements of 30 Urania computed, the elements obtained from JPL HORIZONS and the error in our values.

Table 5: Sun-Earth vector Cartesian components of \mathbf{R} in ecliptic coordinates [Au]

	a	Ω	i	e	ω	τ
	[Au]	[deg]	[deg]		[deg]	[Julian Date]
True	2.365					
Estimated	2.133					
Error	0.232					

We can then determine the orbit of 30 Urania using these orbital elements. The Cartesian components of \mathbf{r} of this asteroid in ecliptic coordinate could be obtained by equation (...), where $\theta = \nu + \omega$:

$$\mathbf{s} = \begin{pmatrix} x \\ y \\ z \end{pmatrix} = r \begin{pmatrix} \cos \Omega \cos \theta - \sin \Omega \cos i \sin \theta \\ \sin \Omega \cos \theta + \cos \Omega \cos i \sin \theta \\ \sin i \sin \theta \end{pmatrix} \quad (24)$$

4 Discussion

First of all, we tried to compute the geocentric position vector of the asteroid, $\rho\mathbf{s}$, and the relative position of the asteroid relative to the Sun, \mathbf{r} , using the equations (...) to (...). In computing the positions, we used the position vector of the Earth relative to the Sun, $\mathbf{R}=(X_{eq}, Y_{eq}, Z_{eq})$, obtained from JPL HORIZONS ephemeris for the given date. Hence, we could determine the orbit of the asteroid using Laplace's method which is explained in section 4 in details. In order to determine the position of the asteroid relative to the Sun, we also needed the right ascension and declination of the asteroid in the course of the four epochs of the observations we had from the DIT files. As it has been mentioned in the report and has been mentioned in full details in my lab 3 report[...], we calculated the plate constants of the CCD used in order to correct for the non-ideal CCD and to obtain more accurate positions of the asteroid on the celestial sphere. However, we faced lots of problems and we were getting ρ to be zero and \mathbf{r} to be almost 1, which meant that the asteroid is 1 Au away from the Sun and 0 Au away from the Earth which meant the asteroid is on the Earth! Therefore, we tried to fix the problem, so we used more accurate Julian dates (with more precision) and also used different epochs. For instance, we used the first, second and third epoch and then we used second, third and fourth epoch to compute ρ and \mathbf{r} , but it did not solve the problem. The other thing we used was to use the coordinates (α and δ) obtained from JPL to check if there is a problem with the code itself or with the coordinates we were using and we found out that the code works perfectly. The values we got for ρ and \mathbf{r} were really close to the true values. The values we had from our conversion using the plate constants were very close to the true values obtained from JPL, they were the same to 4 decimal places; however, as we saw, the calculation of the position vectors is very sensitive to the precision in the α and δ values.

5 Conclusion

6 References

- 1- Lab 4: Astrometric Orbit Determination handout
- 2- Lab 3 Report written by Anita Bahmanyar
- 3-JPL HORIZONS web interface ephemeris

7 Appendix

Here is a it of my code to calculate the mean and eccentric anomaly using equations (...) and (...):

```
for q in range(1,len(t)):
M[q-1]=E[q-1]-e*(np.sin(E[q-1]))
E[q]=E[q-1]+((mu[q-1]-M[q-1])/(1-e*np.cos(E[q-1])))
```

This loops starts from one and continues until it is smaller than the number of elements in time array and calculates the mean anomaly for each time it goes into the loop and using that specific M value, it computes the E value. We should also note that the initial values of $E[0]=\mu[0]$ and $M[0]=E[0]$, where $\mu=n*(t-\tau)$ and t is the time array starting from $\tau=2454868$; therefore, $\mu[0]=0$.

for plotting figure (...), I used the following code:

```
circle_x = a * (np.cos(theta))
circle_y = a * (np.sin(theta))and
r_x = r * (np.cos(theta))
r_y = r * (np.sin(theta))
```

for a circle with radius a and the orbit of Ceres, respectively.

Then I used a for loop to run for 20 times in order to find the ρ and r values close to the true values. This loop goes until the value of r and ρ we have found, converges to its true value:

```
for i in range(20):
rho=((k**2)*((1./(Rlength**3)) - (1./(r1**3)))) * denom
r1 = np.sqrt((rho**2) + (Rlength**2) + (2 * rho * Rot_s))Au
```