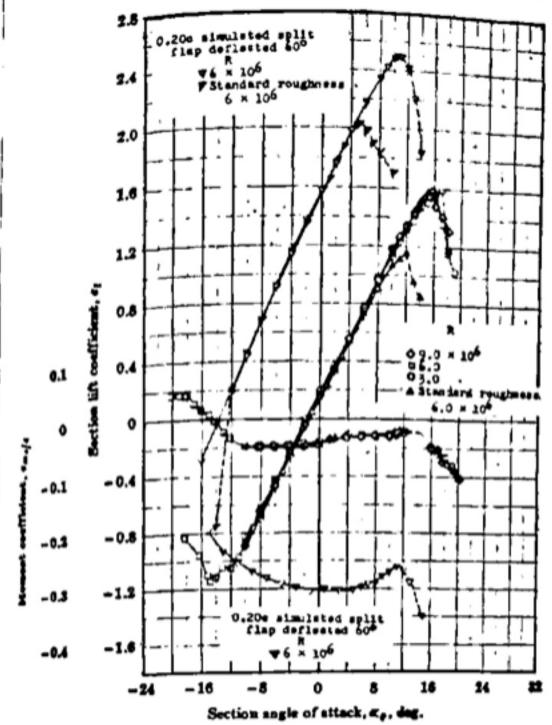


NACA 0012-84 Wing Section, a = 0.8 (modified), $c_{ij} = 0.2$



NACA 0012-84 Wing Section. a = 0.8 (modified), c₄ = 0.2 (Continued)