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MASTER OF SCIENCE THESIS

# STAR-TRACKER PROGRAM FOR CUBESAT SATELLITES

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Warszawa 2017

**Abstract**

Last years directed space industry towards small satellites. Many countries which did not have possibility to enter this branch of industry before now create their own solutions. The goal of this work is to create fully functioning star-tracker software eligible to be used in future satellites as Polish solution of determination of satellite attitude towards Earth. Work contains also description of individual parts and variants of solutions connected with star-tracker.

### **Streszczenie**

Ostatnie lata ukierunkowały przemysł kosmiczny na małe satelity. Wiele krajów, które wcześniej nie miały możliwości wejścia w tę gałąź przemysłu, teraz tworzą własne rozwiązania. Celem niniejszej pracy dyplomowej jest stworzenie w pełni działającego oprogramowania star-trackera nadającego się do wykorzystania w przyszłych satelitach jako polskie rozwiązanie problemu określania orientacji satelity względem ziemi. Praca zawiera też opis poszczególnych części i wariantów rozwiązania problemów związanych ze star-trackerem.

# Contents

<b>Acronyms</b>	<b>6</b>
<b>List of Symbols</b>	<b>7</b>
<b>1 Introduction</b>	<b>8</b>
1.1 Motivation . . . . .	8
1.2 Outline of thesis . . . . .	8
1.3 Related work . . . . .	9
1.4 Cubesat . . . . .	10
1.5 Means of attitude determination . . . . .	11
1.5.1 Inertial Measurement Unit . . . . .	13
1.5.2 Sun Sensors . . . . .	14
1.5.3 Star-Tracker . . . . .	14
1.5.4 Horizon Sensors . . . . .	14
1.5.5 Magnetometer . . . . .	14
1.5.6 Global Positioning System . . . . .	15
1.5.7 Conclusions . . . . .	15
1.6 On-board computer . . . . .	16
<b>2 Preliminaries</b>	<b>17</b>
2.1 Earth's orbits . . . . .	17
2.2 Coordinate frames . . . . .	18
2.2.1 ECI frame . . . . .	18
2.2.2 ECEF frame . . . . .	18
2.2.3 NED frame . . . . .	19
2.2.4 BODY frame . . . . .	19
2.3 Space environment . . . . .	20
2.4 Attitude representations . . . . .	20
2.4.1 Euler angles . . . . .	20
2.4.2 Quaternions . . . . .	21
2.4.3 Advantages of quaternions . . . . .	23
<b>3 Star-tracker program</b>	<b>24</b>
3.1 Centroid - start recognition . . . . .	25
3.2 Star identification . . . . .	27
3.2.1 Angle Matching . . . . .	28
3.2.2 Spherical Triangle Matching . . . . .	29
3.2.3 Planar Triangle . . . . .	30
3.2.4 Pyramid . . . . .	32

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3.2.5	Rate Matching . . . . .	32
3.2.6	Voting . . . . .	32
3.2.7	Grid . . . . .	32
3.3	Star-catalogue and searching for matching stars . . . . .	33
3.3.1	Star Catalogue Generation . . . . .	33
3.3.2	Candidate Matching . . . . .	33
3.3.3	Result Verification . . . . .	33
3.3.4	k-vector . . . . .	33
3.4	Attitude Determination . . . . .	34
3.4.1	q-method . . . . .	35
3.4.2	Wahba's problem . . . . .	36
3.4.3	QUEST . . . . .	37
3.4.4	TRIAD . . . . .	37
3.4.5	The Fast Optimal Attitude Matrix . . . . .	37
3.4.6	DCM (Direction Cosine Matrix) - Singular Value De- composition? . . . . .	37
<b>4</b>	<b>Prototype</b>	<b>39</b>
<b>5</b>	<b>Complete program</b>	<b>40</b>
<b>6</b>	<b>Testing of star-tracker</b>	<b>41</b>
<b>References</b>		<b>42</b>
<b>List of Tables</b>		<b>49</b>
<b>List of Figures</b>		<b>50</b>

## Acronyms

**ADCS** Attitude Determination and Control System.

**GPS** Global Positioning System.

**LEO** Low Earth Orbit.

**LIS** Lost-In-Space.

## List of Symbols

$\phi$  Euler angle, roll.

$\theta$  Euler angle, pitch.

$\psi$  Euler angle, yaw.

$\mathbf{R}(\cdot)$  Rotation matrix using Euler angles.

$\mathbf{q}$  Unit quaternion.

$q_0$  Scalar part of unit quaternion.

$\mathbf{q}_{vec}$  Vector part of unit quaternion.

$\mathbf{Q}$  Quaternion matrix.

$v$  General Euler angle.

$\mathbf{n}$  Unit vector.

$\mathbf{I}$  Identity matrix.

$\mathbf{S}(\cdot)$  Skew symmetric matrix.

$\mathbf{R}_n^b$  Rotation matrix representing a rotation from n to b.

$\mathbf{M}$  Least squares estimate of rotation matrix.

$\mathbf{r}$  Known directional unit vector in the NED frame.

$\mathbf{b}$  Known directional unit vector in the BODY frame.

# 1 Introduction

Stars were used for navigation already ages ago. Together with technological development, spread of sailing and seafaring people started to use more advanced tools helping to more precisely estimate position of ships on sea. In XVIII century new tool was created, named sextant. This device helped to quite precisely calculate angle between the line ship-horizon and ship-star, what let to estimate position.

Nowadays, hundreds of years later, thanks to other technologies like for example Global Positioning System (GPS) stars are not necessary any more for travelling. As it is commonly known, GPS technology is based on satellites, and today those satellites need stars for determining their attitude towards Earth.

With the miniaturization of electronics and batteries new type of satellites was invented: CubeSat micro-satellite. They have now greater sensory and processing power possibilities, previously found only in larger satellites. However CubSats are still behind in terms of attitude determination.

## 1.1 Motivation

The goal of this work is to make fully operational star-tracker software, that could be used on Cubesat satellites. Such program could be used on space missions and could start Polish state-of-the-art technology in growing space technology sector. There is already existing prototype of on-board computer for such CubeSat satellite, which together with this work should create full star-tracker device: hardware + software.

## 1.2 Outline of thesis

This thesis consists of several chapters. Here they are shortly summarized:

**Chapter 1** serves as introduction to this thesis and describes the motivation and goal of this work. It also describes the background of the topic.

**Chapter 2** describes all the important foundations for the fully understanding given work.

**Chapter 3** is the main part of this thesis. It describes how the star-tracker program works and goes through detailed comparison of different approaches.

**Chapter 4** describes the created prototype of star-tracker in Python language.

**Chapter 5** talks about the implementation of star-tracker on the existing prototype of on-board computer.

**Chapter 6** describes how the finished program is performing.

**Chapter 7** contains conclusions about this work and created star-tracker program.

### 1.3 Related work

There exists a number of works connected to topic of star-tracker. Many works are cited later in this thesis as they describe important parts of algorithms. There are just a few works dealing with whole start-tracker program, not just parts of it, and nearly no works about whole program together with implementation on the real equipment.

- generally about CubeSats: [1], [2]
- Star centroids: [3], [4], [5], [6], Lindh [7], Zhang et al. [8],
- Star identification: Geometric Voting [9], Grid Algorithm Padgett and Kreutz-Delgado [10], Brightness Independent [11], Planar Triangle [12], Pyramid [13], Spherical Triangles [14], SP-Search [15], Neural Networks [16], [17] and [18] with separately discussed k-vector algorithm [19], [20], [21], [22], Neural Feruglio et al. [23]
- Attitude Determination: AIM (Attitude estimation using Image Matching) [24], QUEST [25], QUEST improvement [26], Extended Quest [27], EQUEST [28], Singular Value Decomposition Juang et al. [29], summarized

description of attitude estimation algorithms: [30], [31],

Delabie [24]

- hardware [6], Gąska [32], Felikson et al. [33]?

- full star tracker but software only/simulations Kandiyil [34], Huffman et al. [35], Diaz [36]

- full star-tracker with device/on real device Jalabert et al. [37], Lizy-Destrez and Mimoun [38], Rose [39], Mortari and Romoli [40] Cannata et al. [41]

## 1.4 Cubesat

In recent years there has been the development of micro-satellites called CubeSats. CubeSat is a standard created in 1999 at the California Polytechnic State University [2], which is used for low-cost micro-satellites. CubeSats are measured in units. Most are 1U, 2U and 3U. CubeSat 1U has a size of 10 cm on each edge and the maximum weight of 1.333 kg, while the CubeSat 2U is 20x10x10 cm and can weigh up to 2.666 kg.

The advantage of the standard is primarily reduction of the price of elevation of such satellites into orbit. Their popularity is evidenced by the fact that the percentage of satellites weighing less than 10 kg has increased to about 60% of all satellites, and only CubeSats make up about half of the small satellites that were launched in last years [1][42]. Till now there were around 800 CubeSats launched [43].

There exist wide range of CubeSat examples. Mostly they are scientific and students' satellites, for learning, experimental and science purposes. There are also few Polish examples: PW-SAT (1U) and PW-SAT 2 (2U) designed by students of Warsaw University of Technology, and Lem (8U) and Heweliusz (8U), both made at Polish Academy of Sciences. PW-SAT and PW-SAT 2 were used for learning and experimenting with new technologies, like deorbitation sail, solar sensor, etc. [44]. Scientific satellites Lem and Heweliusz were built as part of BRITE programme - joint programme of Austria, Canada and Poland. The goal of this programme is to research mechanisms of energy transportation and angular momentum which happen

inside hottest stars [45].

Typical CubeSat consist of many different components. Quite often it has antennas and radios for communication with mission centre on Earth, Electric Power System (EPS - subsystem responsible for managing electric power in the satellite), Attitude Determination and Control (ADCS - subsystem responsible for analysing data from sensors and taking decision about trajectory, etc.), magnetometer, star-tracker, sun sensors, payload processor with mission instructions, etc. CubeSats are quite small, hence they usually do not have any propulsion system. Figure 2 shows the components of CubeSat on example of NASA's Interplanetary NanoSpacecraft Pathfinder In a Relevant Environment (INSPIRE).

Costs of such CubeSat satellite on example of PW-SAT 2: 120,000-200,000 EUR for putting the satellite on the orbit and round 70,000 EUR for designing and building the satellite itself [49].

## 1.5 Means of attitude determination

Attitude of spaceships must be usually stabilised and controlled for various reasons. It is necessary for satellite antenna to be pointing towards Earth for the proper communication, to intelligently control the heat by using the effects of cooling and heating of the shadows and sunlight, as well as to navigate: manoeuvres must be performed in the right direction.

Attitude is determined between two coordinate systems (where one is reference system) and defines by what angles the coordinate system connected with the researched object has to be shifted in order to cover the reference system. Devices such as planes and satellites have so called Attitude Determination and Control System (ADCS), which controls attitude of object relative to an inertial reference frame or another entity (the celestial sphere, certain areas, the nearby objects, etc.).

Currently, attitude determination of CubeSats is limited mainly to the sun sensors, magnetometers and measurements of inertia. The following table describes the accuracy of different sensors. Unquestionable winner here is the star-tracker, which, due to its quality and uniqueness of the constellations is ideal for navigation.

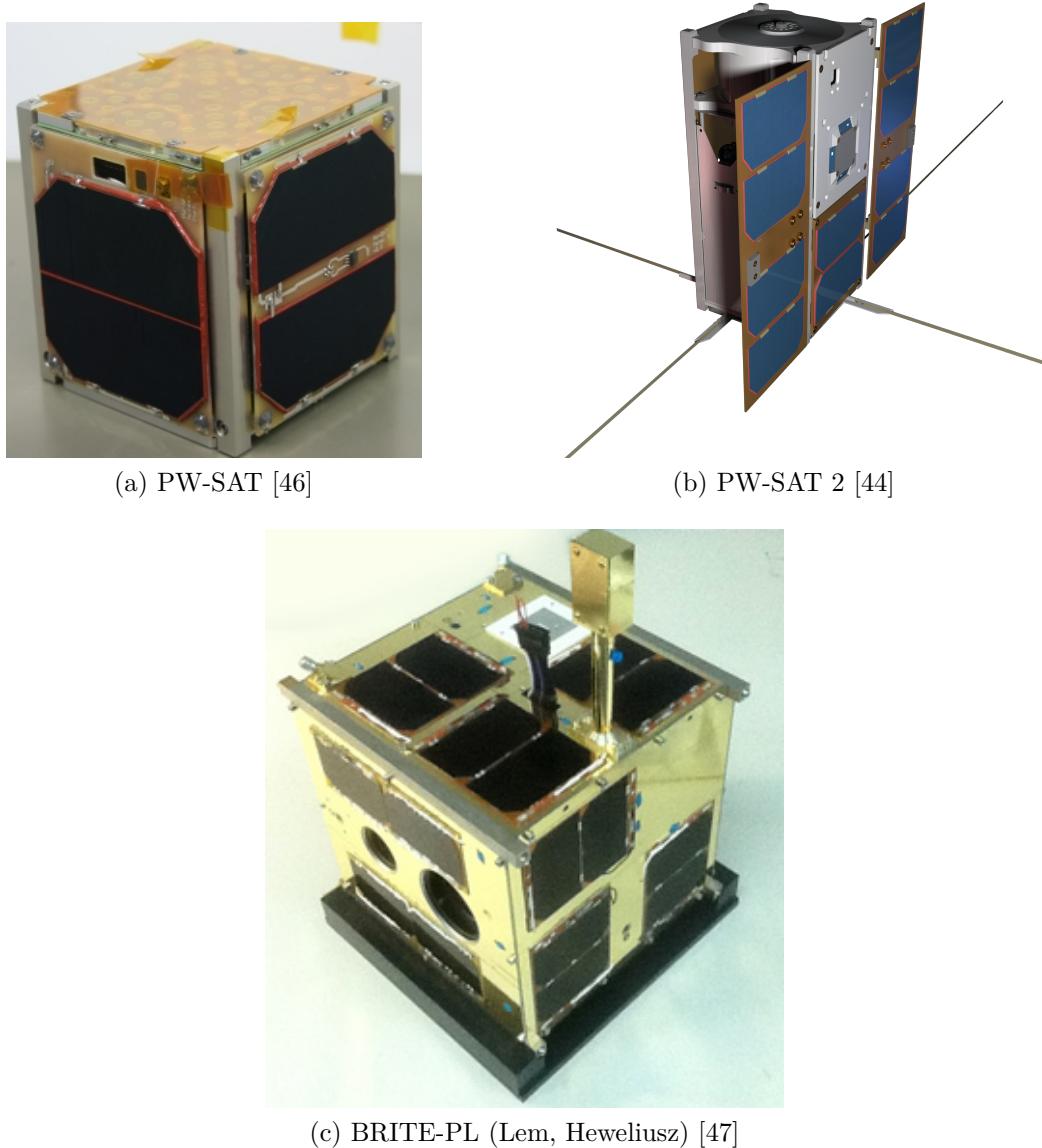


Figure 1: Example of Polish CubeSats.

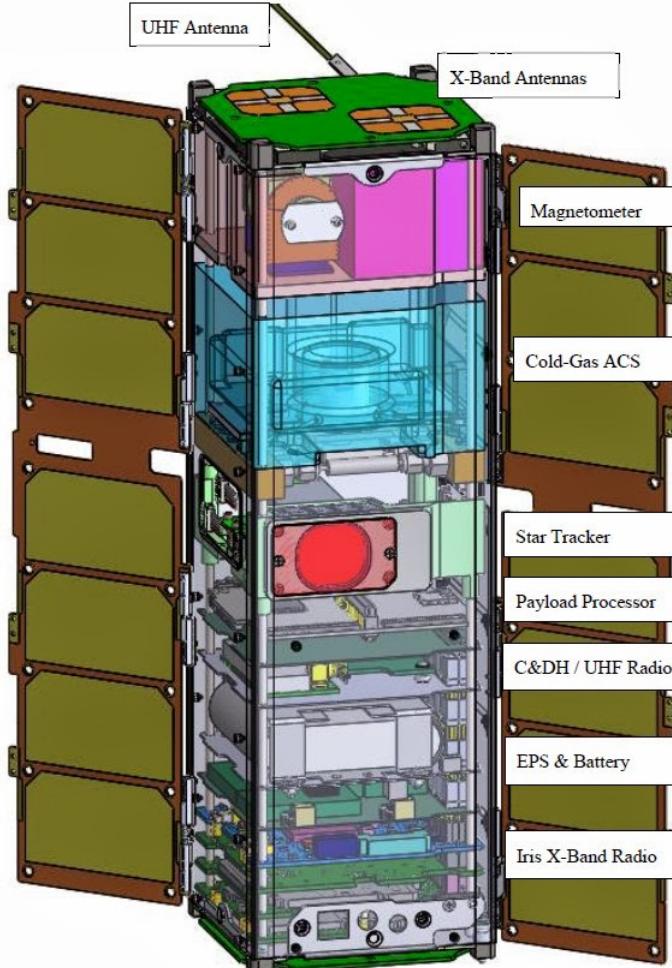


Figure 2: CubeSat build - INSPIRE, Image [48]

### 1.5.1 Inertial Measurement Unit

An Inertial Measurement Unit (IMU) detects linear acceleration using one or more accelerometers and rotational rate using gyroscopes, sometimes they also include magnetometers. They are commonly used in planes, spacecrafts and many consumer devices like smart-phones. The biggest disadvantage of IMU is accumulated error. Even small error that is accumulated with time brings device to think it is in different location than it actually is. Usually the errors are corrected via use of Kalman filter, which corrects IMU errors using other sources (i.e. GPS).

### 1.5.2 Sun Sensors

Sun sensors typically consist of few sensors oriented perpendicularly, which can tell the angle of the sun. They can give the information how the satellite is oriented towards the sun.

### 1.5.3 Star-Tracker

Star-tracker is quite sophisticated device, which consists of camera, processor and memory for database. In the database is catalogue of stars, built basing on data from Earth's observatories. Simply put star-tracker, when in orbit, takes picture of space - stars. Then processes the image to recognize stars in it and compares with data in catalogue or with previous image. When stars are identified star-tracker estimates the attitude of satellite towards Earth. The main advantage over other means: great accuracy and star-tracker can find its location again after satellite was restarted or lost in space.

### 1.5.4 Horizon Sensors

Earth horizon sensors use infra-red radiation from the Earth's surface via use of bolometer. The device detects when infra-red signal was first detected and then lost. The time between is used to determine the Earth's width, what later can be used to determine roll angle. In other words device detects the contrast between the cold of deep space and the heat of Earth's atmosphere. Problems of this sensors are the fact that Earth is not perfectly circular and sensors detect infra-red in the atmosphere, which varies depending on the latitude.

### 1.5.5 Magnetometer

The device actually consists of three magnetometers - one for measuring Earth's magnetic field in each direction. Measurements can be compared with known magnetic field given by International Geomagnetic Reference Field model. The problem with this sensors is that they are susceptible to noise and outside of Earth's orbit it is completely unpredictable.

### 1.5.6 Global Positioning System

Nowadays nearly everyone has personal device with Global Positioning System (GPS) navigation. It is also possible to use GPS satellites to determine spacecraft's attitude. This is possible when at least four GPS satellites are visible by spacecraft, what provides three-dimensional position. Even though it gives quite good accuracy, but it decreases with the altitude. GPS satellites are located on Medium Earth Orbit (MEO), around 20,200 km above sea [50]. It is possible to navigate spacecraft between Low Earth orbit (LEO - 160 to 2,000 km) and Geosynchronous Orbit (GEO - 35,786 km) [51]. For missions beyond Earth's orbit GPS navigation is however not useful. Example of GPS usage for navigation in space is International Space Station (ISS) [52]. There are however works that claim it could make using GPS systems possible for attitude determination up to Moon attitude in the future [53].

### 1.5.7 Conclusions

The above means of attitude determination are usually used together with the use of Kalman filter, to compensate errors of one sensor with the other. However none of the means is more accurate than star-trackers. Sun sensors can provide very accurate measurements, but can only operate in sunlight. For low-Earth orbit (LEO), even 30% of the orbit can be done in the darkness. Magnetometers are small and can give accurate measurements if properly calibrated. Their drawback is the limited knowledge of the magnetic field and electromagnetic interference due to highly integrated construction of CubeSats. Microelectromechanical gyroscopes are small enough to fit into a CubeSats. However, they suffer from sudden movements, and could not maintain the correct measurement of the 15-minute period of the eclipse orbit LEO. GPS navigation is accurate, however only on Earth's orbit. To be truly competitive and reliable platform, CubeSats must provide the correct determination of attitude. The best way to meet this goal is via using star-trackers. The typical accuracies of different sensors are available in Table 1.

Sensor	Accuracy (less is better)
Inertial Measurement Unit	0.001°/hr to 1°/hr
Sun Sensors	0.005° to 3°
Star Sensors	0.0003° to 0.01° (1 arc sec to 1 arc min)
Horizon Sensors	0.05° (GEO) 0.1° (LEO)
Magnetometer	1.0° (5000km alt) 5.0° (200 km alt)

Table 1: Sensor Accuracy Ranges. Adapted from Hall [31] and Larson and Wertz [54]

## 1.6 On-board computer

This section will describe the on-board computer which was done as part of other thesis.

## 2 Preliminaries

### 2.1 Earth's orbits

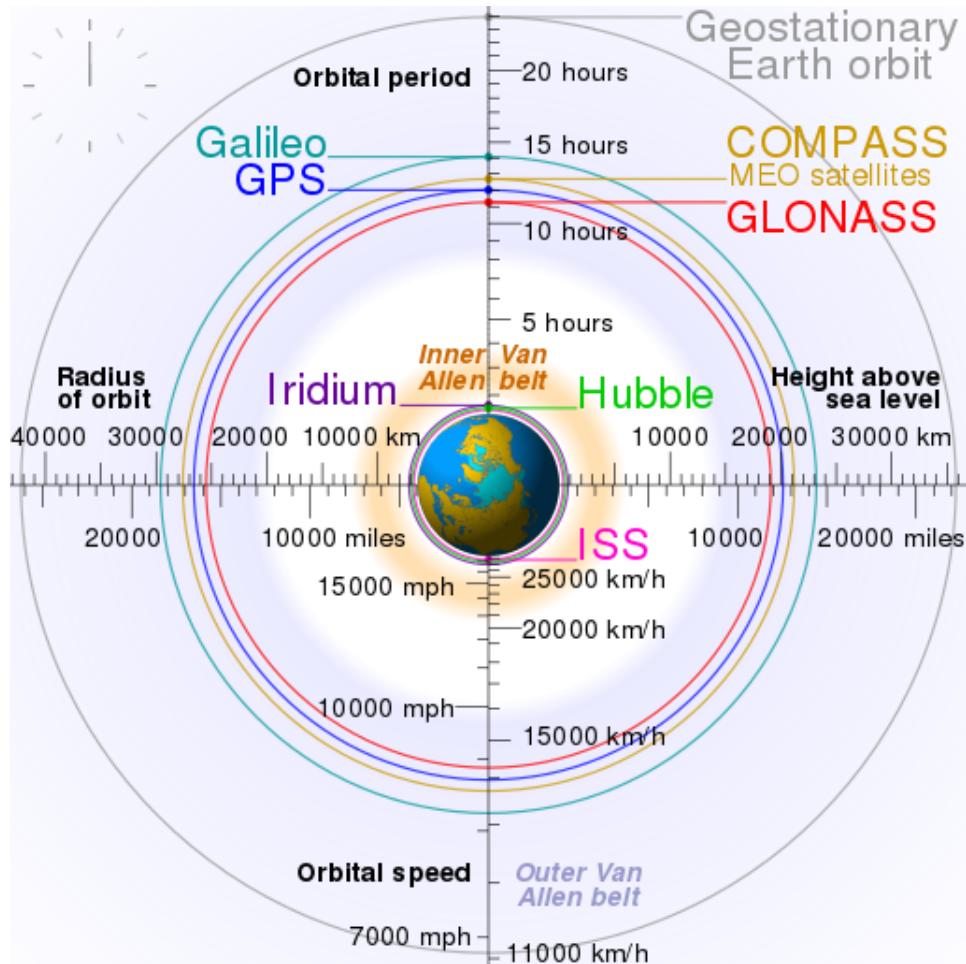


Figure 3: Earth's satellite navigation orbits [55]

By altitude:

Low Earth orbit (LEO): geocentric orbits with altitudes from 160 to 2,000 km (100–1,240 miles).[2]

Medium Earth orbit (MEO): geocentric orbits ranging in altitude from 2,000 km (1,240 miles) to just below geosynchronous orbit at 35,786 kilometers (22,236 mi). Also known as an intermediate circular orbit. These

are "most commonly at 20,200 kilometers (12,600 mi), or 20,650 kilometers (12,830 mi), with an orbital period of 12 hours."<sup>[3]</sup>

Both geosynchronous orbit (GSO) and geostationary orbit (GEO) are orbits around Earth matching Earth's sidereal rotation period. All geosynchronous and geostationary orbits have a semi-major axis of 42,164 km (26,199 mi).<sup>[4]</sup> This works out to an altitude of 35,786 km (22,236 mi). All geostationary orbits are also geosynchronous, but not all geosynchronous orbits are geostationary. A geostationary orbit stays exactly above the equator, whereas a geosynchronous orbit may swing north and south to cover more of the Earth's surface. Both complete one full orbit of Earth per sidereal day (relative to the stars, not the Sun).

High Earth orbit: geocentric orbits above the altitude of geosynchronous orbit 35,786 km (22,240 miles).<sup>[3]</sup>

By

## 2.2 Coordinate frames

### 2.2.1 ECI frame

The Earth Centered Inertial frame has its x-axis pointing towards the vernal equinox, and its z-axis pointing along the rotation axis of the Earth at some initial time. The y-axis completes a right handed orthogonal coordinate system. The frame's origin is at the center of the Earth. Larson and Wertz [54]

### 2.2.2 ECEF frame

This frame also has its origin at the center of the Earth, but the Earth Centered Earth Fixed frame has its x-axis pointing towards the point where the intersection between the longitude and latitude have zero value. It can also be described as the intersection between the Greenwich meridian and the Equator. The frame's z-axis is pointing along the Earth's rotation axis. The y-axis completes the right handed orthogonal system. The ECEF frame

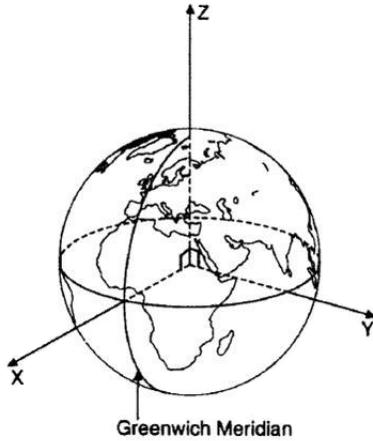


Figure 4: ECI frame, Image Larson and Wertz [54]

is not an inertial frame, it rotates relative to the ECI frame along the Earth rotation.

### 2.2.3 NED frame

The North East Down frame has its z-axis pointing downwards, perpendicular to the tangent plane of the Earth's reference ellipsoid. The ellipsoid is mathematically defined and fitted for approximation of the Earth. The x-axis points towards true north and the y-axis points East. The NED frame is an inertial frame.

### 2.2.4 BODY frame

This frame is attached to the satellite, and is moving and rotating with it. The origin coincides with the origin of the NED frame. The axes coincide with the principle axes of inertia; the x-axis is pointing forwards, the y-axis is pointing to the right side and the z-axis is pointing downwards through the camera side of the satellite.

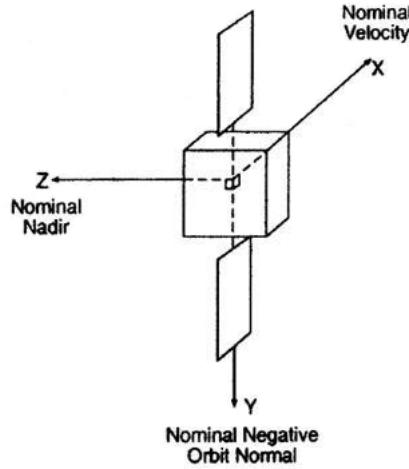


Figure 5: BODY frame, Image Larson and Wertz [54]

## 2.3 Space environment

## 2.4 Attitude representations

Several representations for describing attitude are available, the most common being Euler angles. More complicated attitude representations are quaternions. Quaternions are used for all the estimation methods presented in this thesis. They are singular-free, and are therefore well suited for attitude determination.

### 2.4.1 Euler angles

Euler angles were first described by Leonhard Euler in 1776, and are used to represent the orientation of a body Euler [56]. Three parameters are required for a full understanding of the orientation between two frames, one angle for the rotation around each of the axes. The angles are called roll, pitch and yaw and are usually written as phi, rho and psi. The Euler angles are often used for the definition of rotation matrices about the x, y and z-axis. In R<sub>3</sub>, the coordinate system rotations in a counter-clockwise direction looking towards the origin are given from:

$$\mathbf{R}_x(\phi) = \begin{bmatrix} 1 & 0 & 0 \\ 0 & \cos(\phi) & -\sin(\phi) \\ 0 & \sin(\phi) & \cos(\phi) \end{bmatrix} \quad (1)$$

$$\mathbf{R}_y(\theta) = \begin{bmatrix} \cos(\theta) & 0 & \sin(\theta) \\ 0 & 1 & 0 \\ -\sin(\theta) & 0 & \cos(\theta) \end{bmatrix} \quad (2)$$

$$\mathbf{R}_z(\psi) = \begin{bmatrix} \cos(\psi) & -\sin(\psi) & 0 \\ \sin(\psi) & \cos(\psi) & 0 \\ 0 & 0 & 1 \end{bmatrix} \quad (3)$$

#### 2.4.2 Quaternions

Hamilton [57] Cayley [58] Courant and Hilbert [59] Mebius [60] Barile [61] Shoemake [62] Horn [63]

Quaternions were first described by Sir William Rowan Hamilton in 1843 Hamilton [57]. His intention was to find an extension of vector algebra, and in 1845 Arthur Cayley published an article where he used multiplication of quaternions to describe rotations Cayley [58]. Three of the four elements of a quaternion give the coordinates for the axis of rotation, while the fourth is described by the angle of rotation Courant and Hilbert [59]. A quaternion can be written as a four-dimensional vector:

$$\mathbf{q} := \begin{bmatrix} q_0 \\ q_1 \\ q_2 \\ q_3 \end{bmatrix} \quad (4)$$

The real part of the quaternion behaves like a scalar in the three-dimensional vector space. Using a rotation angle  $v_i$ , the real part can be written as:

$$q_0 = \cos(v/2) \quad (5)$$

$$\mathbf{n} = \frac{\mathbf{n}}{||\mathbf{n}||} \quad (6)$$

The imaginary part uses a unit vector given from  $\mathbf{n} = \mathbf{n}/\|\mathbf{n}\|$  (wzór) where the norm of the vector,  $\|\mathbf{n}\|$ , is defined as the square root of each of the squared elements of  $\mathbf{n}$  added together. Throughout the thesis, vectors and matrices will be written in bold print. The imaginary part can be written as a vector:

$$\mathbf{q}_{vec} := \begin{bmatrix} q_1 \\ q_2 \\ q_3 \end{bmatrix} = [\mathbf{n} \sin(v/2)] \quad (7)$$

There are several ways to write quaternions. Sometimes it is convenient to think of a quaternion as the sum of a scalar and a vector written as:

$$\mathbf{q} := q_0 + \mathbf{q}_{vec} = q_0 + q_1 i + q_2 j + q_3 k \quad (8)$$

Complex numbers can be represented as matrices, and so can quaternions. A quaternion describes a point in 4D space, and can be represented by a  $4 \times 4$  matrix by using a left-isoclinic rotation as proved in Mebius [60], and used in Barile [61] and Shoemake [62]:

$$\mathbf{Q} = \begin{bmatrix} q_0 & -q_1 & -q_2 & -q_3 \\ q_1 & q_0 & -q_3 & q_2 \\ q_2 & q_3 & q_0 & -q_1 \\ q_3 & -q_2 & q_1 & q_0 \end{bmatrix} \quad (9)$$

The transpose of the matrix is the same as the conjugate of the quaternion:

$$\mathbf{q}^* := q_0 - \mathbf{q}_{vec} = q_0 + q_1 i + q_2 j + q_3 k \quad (10)$$

Two quaternions are conjugate if they are orthogonal with respect to their inner product Horn [63]. The inverse of a quaternion  $\mathbf{q}$  is defined as  $\mathbf{q}^{-1} = 1/\mathbf{q}^*$ . In this report, the mathematics are based on unit quaternions, which satisfies the constraint:

$$\mathbf{q}^T \mathbf{q} = 1 \quad (11)$$

The length of a unit quaternion is 1, which leads its inverse to be its conjugate.

### 2.4.3 Advantages of quaternions

The unit quaternion notation is compact, and round off errors are easier to handle than for matrix representation. The nearest orthonormal matrix to one that is not quite orthonormal, is difficult to find. Multiplying unit quaternions may similarly lead to quaternions that are no longer of unit length, but these can easily be normalized to make sure they correspond to valid rotations. The computational cost of normalizing a quaternion is much less than for normalizing a matrix. Quaternions are safe from a phenomenon called gimbal lock. When the pitch angle in a pitch/roll/yaw-system is rotated 90° up or down, and the yaw and roll correspond to the same motion, a degree of freedom of rotation can be lost. In a gimbal-based aerospace inertial navigation system, this could have disastrous results if the aircraft is in a steep dive or ascent Shoemake [62]. The quaternion elements vary continuously over the unit sphere in  $R^4$ , (denoted by  $S^3$ ) as the orientation changes, avoiding this problem. Due to the possible spin in the CubeSat, combined with the singularity problem for Euler angles, an attitude estimation method based on unit quaternions is preferred.

### 3 Star-tracker program

Star-tracker program is designed to determine the attitude of the satellite with the image of the stars made by the satellite camera. Before the start of the mission star catalogue is generated, based on the star catalogues obtained from the observatories on Earth and uploaded to the computer on-board the satellite.

After the start, the satellite is released from rocket's tank in space and has no idea where it is. Then the star-tracker on the satellite enters into Lost-In-Space mode (LIS) and analyses the current image of the stars of the camera, and then searches the corresponding result of stars in the on-board star catalogue database. If it finds an entry in the database, the satellite goes into tracking mode. This means each next calculation of the orientation of the satellite attitude is going to be based on a comparison of the current pictures of stars with the preceding ones. If it cannot find the result in the database, this action is repeated from time to time, until a match is found in the database and the program will go into tracking mode.

Of course LIS does not happen only at the beginning of the satellite's flight, but can also result from many causes, e.g. a satellite which is for a long time in the darkness may discharge its battery, and during the next entry into sunlit zone will turn on and look again its attitude. Another case is when satellite becomes lost, although once found the orientation and follow her. This happens, because the successive results are based on preceding ones and even the smallest mistake will grow until the satellite will not be able to correctly calculate their attitude. It may also happen that the satellite will rotate around its axis so fast that the program would not keep up with the processing the image and calculations. In this case, the corresponding other satellite's systems should take an action reducing his spin, but this is not part of this work. It is showed in simple conceptual diagram - Figure 6.

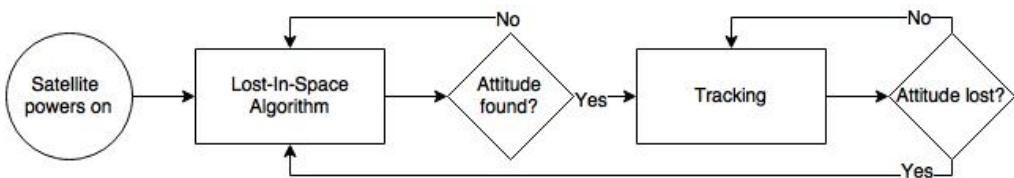


Figure 6: Startracker conceptual algorithm diagram

Each part - LIS and tracking - consists of smaller algorithms. Lost-In-

Space at the beginning of reading an image from the camera, thresholds the image. Thresholding is basically throwing away parts of the image which do not exceed the threshold brightness, making just bright enough stars taken into account. Next it calculates centroid (center of mass) of selected stars, identifies these stars using one of the possible methods (here Planar Triangle, described later) and searches the directory on-board technology (k-vector). If the database does not contain corresponding stars, the algorithm returns to the input state and starts analysing the next image. If it manages to find the corresponding stars, the program determines the attitude of the satellite (by what angle is satellite shifted comparing to the referenced object - Earth or previous image) and enters tracking mode. Tracking mode is in large part similar to the LIS. The algorithm also first analyses the photo, selects and calculates centroids stars, identifies them, but does not search for stars in the directory board. In this algorithm are compared two images, one following after the other, and on this basis the current attitude is calculated.

Generally star-tracker is divided into three main parts McBryde and Lightsey [64]:

- recognising stars on the image and converting the data into list of star vectors by calculating star centroids;
- identifying which star vector represents which real star in catalogue. This is done by comparing star vectors from the image with data in star catalogue, which is generated before space mission;
- estimating the attitude by calculating the displacement between two frames.

### 3.1 Centroid - start recognition

Samaan et al. [4]

Liebe [3]

The whole star-tracker program is based on very precise calculations. For this reason, the calculation based on the position of the pixels only can give incorrect results. It is necessary to calculate the position of stars with an

accuracy exceeding pixels. This is why the calculation of the star centroids is necessary.

The first step is the determination of stars' position in the plane of the image. If focused star pictures are recorded, the image of each star will fall only on one or two pixels, and most likely the saturate these pixels, resulting in a pixel level accuracy. Many star-trackers are doing deliberately blurred images, in order to spread the photons on a larger number of pixels, which allows the algorithm for calculating the centroid in sub-pixel accuracy.

After the registration of such image, the centroid of the star is found similarly to the centroid weight points array, with a few differences. Firstly, instead of weight light intensity is used. Secondly, the intensity of light is usually normalized by the pixels around the star in order to filter out glare or noise. The resulting outcome is a series of two-dimensional coordinates on the photo plane with the starting point at the center of the image. This allows the system to the coordinates of the stars can be easily converted into unit vectors in the next step.

The algorithm requires specification of the light intensity threshold (to select the brightest stars)  $I_{thresh}$  and the size of the Region of Interest (ROI)  $a_{ROI}$  in pixels. These values are adjusted to manipulate the performance of the algorithm. For example, the higher the value of  $I_{thresh}$  is more resistant to noise, but can miss some real stars in the picture. Similarly, a large value  $a_{ROI}$  means a more exact value of centroid, but the algorithm can see one star there, where are actually two within a short distance of each other. The centroiding part is about trade-off between noise resistance and star recognition, and also between recognizing few close stars as one and recognizing one big star as a few. Please note that  $a_{ROI}$  must be a positive odd number for the proper functioning of the algorithm.

The idea of how to calculate such centroids is adapted from McBryde and Lightsey [64] and described below:

$$x_{start} = x - \frac{a_{ROI} - 1}{2} \quad (12)$$

$$y_{start} = y - \frac{a_{ROI} - 1}{2} \quad (13)$$

$$x_{end} = x_{start} + a_{ROI} \quad (14)$$

$$y_{end} = y_{start} + a_{ROI} \quad (15)$$

$$I_{bottom} = \sum_{i=1}^{x_{end}-1} I(i, y_{start}) \quad (16a)$$

$$I_{top} = \sum_{i=2}^{x_{end}} I(i, y_{end}) \quad (16b)$$

$$I_{left} = \sum_{j=1}^{y_{end}-1} I(x_{start}, j) \quad (16c)$$

$$I_{right} = \sum_{j=2}^{y_{end}} I(x_{start}, j) \quad (16d)$$

$$I_{border} = \frac{I_{top} + I_{bottom} + I_{left} + I_{right}}{4(a_{ROI} - 1)} \quad (16e)$$

$$\tilde{I}(x, y) = I(x, y) - I_{border} \quad (17)$$

$$B = \sum_{i=x_{start}+1}^{x_{end}-1} \sum_{j=y_{start}+1}^{y_{end}-1} \tilde{I}(i, j) \quad (18)$$

$$x_{CM} = \sum_{i=x_{start}+1}^{x_{end}-1} \sum_{j=y_{start}+1}^{y_{end}-1} \frac{i \times \tilde{I}(i, j)}{B} \quad (19)$$

$$y_{CM} = \sum_{i=x_{start}+1}^{x_{end}-1} \sum_{j=y_{start}+1}^{y_{end}-1} \frac{j \times \tilde{I}(i, j)}{B} \quad (20)$$

$$u = \frac{\begin{bmatrix} \mu x_{CM} & \mu y_{CM} & f \end{bmatrix}^T}{\left\| \begin{bmatrix} \mu x_{CM} & \mu y_{CM} & f \end{bmatrix} \right\|} \quad (21)$$

### 3.2 Star identification

all Spratling and Mortari [65]

Brightness Independent 4-Star Matching Algorithm for Lost-in-Space 3-Axis Attitude Acquisition Dong et al. [11]

SP-Search: A New Algorithm for Star Pattern Recognition Mortari [15]

Star Identification using Neural networks Miri and Shiri [16] Lindblad et al. [17]

Star pattern recognition using neural networks Li et al. [18]

### 3.2.1 Angle Matching

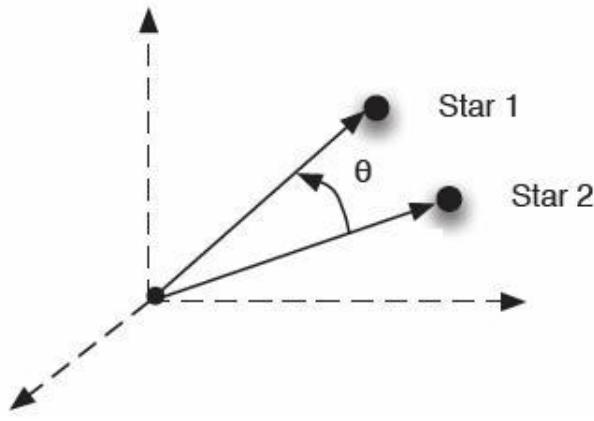


Figure 7: Vector angle method, Image Gottlieb [66]

Gottlieb [66]

$$\theta = \cos^{-1}(\mathbf{r}_1 \cdot \mathbf{r}_2) \quad (22)$$

$$\tilde{\mathbf{b}}_i = A\mathbf{r}_i \quad (23)$$

$$\tilde{\mathbf{b}}_i = A\mathbf{r}_i + \mathbf{v}_i, \quad \mathbf{v}_i^T A\mathbf{r}_i = 0 \quad (24)$$

$$E \{ \mathbf{v}_i \} = 0 \quad (25a)$$

$$E \{ \mathbf{v}_i \mathbf{v}_i^T \} = \sigma_i^2 [\mathbf{I} - (A\mathbf{r}_i)(A\mathbf{r}_i)^T] \quad (25b)$$

$$\mathbf{b}_i^T \mathbf{b}_j = \mathbf{r}_i^T A^T A \mathbf{r}_j = \mathbf{r}_i^T \mathbf{r}_j \quad (26)$$

$$\begin{aligned} \tilde{\mathbf{b}}_i &= A\mathbf{r}_i + \mathbf{v}_i \\ \tilde{\mathbf{b}}_j &= A\mathbf{r}_j + \mathbf{v}_j \end{aligned}$$

$$z \equiv \tilde{\mathbf{b}}_i^T \tilde{\mathbf{b}}_j = \mathbf{r}_i^T \mathbf{r}_j + \mathbf{r}_i^T A^T \mathbf{v}_j + \mathbf{r}_j^T A^T \mathbf{v}_i + \mathbf{v}_i^T \mathbf{v}_j \quad (28)$$

$$E \{ z \} = \mathbf{r}_i^T \mathbf{r}_j \quad (29)$$

$$p \equiv z - E \{ z \} = \mathbf{r}_i^T A^T \mathbf{v}_j + \mathbf{r}_j^T A^T \mathbf{v}_i + \mathbf{v}_i^T \mathbf{v}_j \quad (30)$$

$$\begin{aligned} \sigma_p^2 \equiv E \{ p \} = \\ \mathbf{r}_1^T A^T R_2 A \mathbf{r}_1 + \mathbf{r}_2^T A^T R_a A \mathbf{r}_2 + \text{Trace}(R_1 R_2) = \\ \text{Trace}(A \mathbf{r}_1 \mathbf{r}_1^T R_2) + \text{Trace}(A \mathbf{r}_2 \mathbf{r}_2^T R_1) + \text{Trace}(R_1 R_2) \end{aligned} \quad (31)$$

### 3.2.2 Spherical Triangle Matching

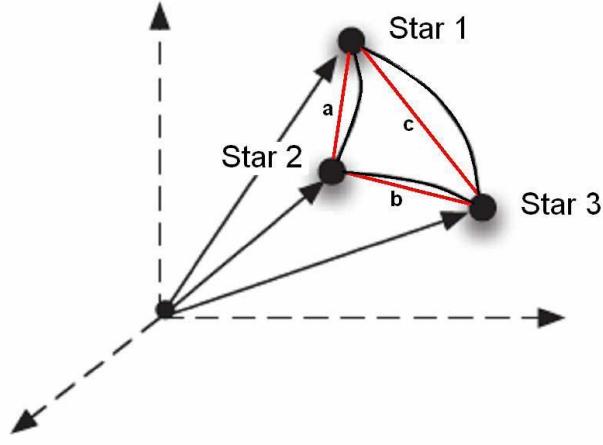


Figure 8: Spherical Triangle Method, Image Cole and Crassidus [14]

Cole and Crassidus [14]

$$A = 4 \tan^{-1} \sqrt{\tan \frac{s}{2} \tan \frac{s-a}{2} \tan \frac{s-b}{2} \tan \frac{s-c}{2}} \quad (32)$$

$$\begin{aligned} s &= \frac{1}{2}(a+b+c) \\ a &= \cos^{-1} \left( \frac{\mathbf{b}_1 \cdot \mathbf{b}_2}{|\mathbf{b}_1| |\mathbf{b}_2|} \right) \\ b &= \cos^{-1} \left( \frac{\mathbf{b}_2 \cdot \mathbf{b}_3}{|\mathbf{b}_2| |\mathbf{b}_3|} \right) \\ c &= \cos^{-1} \left( \frac{\mathbf{b}_3 \cdot \mathbf{b}_1}{|\mathbf{b}_3| |\mathbf{b}_1|} \right) \end{aligned}$$

$$I_p = \sum \theta^2 dA \quad (34)$$

### 3.2.3 Planar Triangle

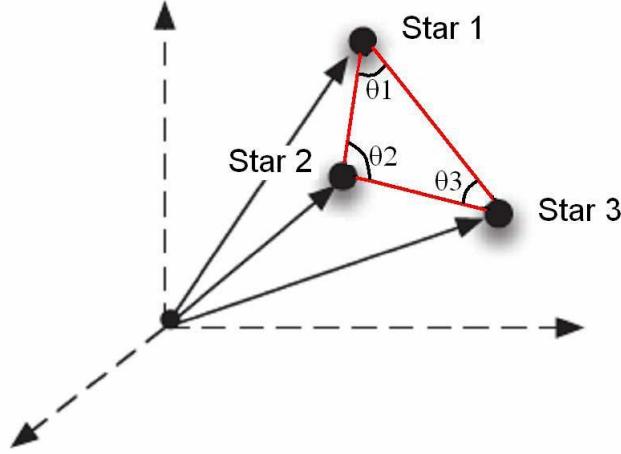


Figure 9: Planar Triangle Method, Image Cole and Crassidis [12]

Cole and Crassidis [12]

$$s = \frac{1}{2}(a + b + c) \quad (35a)$$

$$a = \|\mathbf{u}_p - \mathbf{u}_q\| \quad (35b)$$

$$b = \|\mathbf{u}_q - \mathbf{u}_r\| \quad (35c)$$

$$c = \|\mathbf{u}_p - \mathbf{u}_r\| \quad (35d)$$

$$A = \sqrt{s(s-a)(s-b)(s-c)} \quad (36)$$

$$J = A \frac{(a^2 + b^2 + c^2)}{36} \quad (37)$$

Derivatives

$$H = \begin{bmatrix} \mathbf{h}_1^T & \mathbf{h}_2^T & \mathbf{h}_3^T \end{bmatrix} \quad (38)$$

$$\mathbf{h}_1^T \equiv \frac{\delta A}{\delta a} \frac{\delta a}{\delta \mathbf{b}_1} + \frac{\delta A}{\delta c} \frac{\delta c}{\delta \mathbf{b}_1} \quad (39a)$$

$$\mathbf{h}_2^T \equiv \frac{\delta A}{\delta a} \frac{\delta a}{\delta \mathbf{b}_2} + \frac{\delta A}{\delta b} \frac{\delta b}{\delta \mathbf{b}_2} \quad (39b)$$

$$\mathbf{h}_3^T \equiv \frac{\delta A}{\delta b} \frac{\delta b}{\delta \mathbf{b}_3} + \frac{\delta A}{\delta c} \frac{\delta c}{\delta \mathbf{b}_3} \quad (39c)$$

$$\frac{\delta A}{\delta a} = \frac{u_1 - u_2 + u_3 + u_4}{4A} \quad (40a)$$

$$\frac{\delta A}{\delta b} = \frac{u_1 + u_2 - u_3 + u_4}{4A} \quad (40b)$$

$$\frac{\delta A}{\delta c} = \frac{u_1 + u_2 + u_3 - u_4}{4A} \quad (40c)$$

$$u_1 = (s - a)(s - b)(s - c) \quad (41a)$$

$$u_2 = s(s - b)(s - c) \quad (41b)$$

$$u_3 = s(s - a)(s - c) \quad (41c)$$

$$u_4 = s(s - a)(s - b) \quad (41d)$$

$$\frac{\delta a}{\delta \mathbf{b}_1} = (\mathbf{b}_1 - \mathbf{b}_2)^T / a, \quad \frac{\delta a}{\delta \mathbf{b}_2} = -\frac{\delta a}{\delta \mathbf{b}_1} \quad (42a)$$

$$\frac{\delta b}{\delta \mathbf{b}_2} = (\mathbf{b}_2 - \mathbf{b}_3)^T / b, \quad \frac{\delta b}{\delta \mathbf{b}_3} = -\frac{\delta b}{\delta \mathbf{b}_2} \quad (42b)$$

$$\frac{\delta c}{\delta \mathbf{b}_1} = (\mathbf{b}_1 - \mathbf{b}_3)^T / c, \quad \frac{\delta c}{\delta \mathbf{b}_3} = -\frac{\delta c}{\delta \mathbf{b}_1} \quad (42c)$$

$$\sigma_A^2 = H R H^T \quad (43)$$

$$R \equiv \begin{bmatrix} R_1 & 0_{3x3} & 0_{3x3} \\ 0_{3x3} & R_2 & 0_{3x3} \\ 0_{3x3} & 0_{3x3} & R_3 \end{bmatrix} \quad (44)$$

Polar Moment

$$\bar{H} = \left[ \bar{\mathbf{h}}_1^T \quad \bar{\mathbf{h}}_2^T \quad \bar{\mathbf{h}}_3^T \right] \quad (45)$$

$$\bar{\mathbf{h}}_1^T \equiv \frac{\delta J}{\delta a} \frac{\delta a}{\delta \mathbf{b}_1} + \frac{\delta J}{\delta c} \frac{\delta c}{\delta \mathbf{b}_1} + \frac{\delta J}{\delta A} \mathbf{h}_1^T \quad (46a)$$

$$\bar{\mathbf{h}}_2^T \equiv \frac{\delta J}{\delta a} \frac{\delta a}{\delta \mathbf{b}_2} + \frac{\delta J}{\delta b} \frac{\delta b}{\delta \mathbf{b}_2} + \frac{\delta J}{\delta A} \mathbf{h}_2^T \quad (46b)$$

$$\bar{\mathbf{h}}_3^T \equiv \frac{\delta J}{\delta b} \frac{\delta b}{\delta \mathbf{b}_3} + \frac{\delta J}{\delta c} \frac{\delta c}{\delta \mathbf{b}_3} + \frac{\delta J}{\delta A} \mathbf{h}_3^T \quad (46c)$$

$$\frac{\delta J}{\delta a} = Aa/18, \quad \frac{\delta J}{\delta b} = Ab/18, \quad \frac{\delta J}{\delta c} = Ac/18 \quad (47a)$$

$$\frac{\delta J}{\delta A} = (a^2 + b^2 + c^2)/36 \quad (47b)$$

$$\sigma_J^2 = \bar{H} R \bar{H}^T \quad (48)$$

### 3.2.4 Pyramid

Mortari et al. [13]

### 3.2.5 Rate Matching

Samaan et al. [67] to be removed?

### 3.2.6 Voting

Kolomenkin et al. [9]

### 3.2.7 Grid

Padgett and Kreutz-Delgado [10]

### 3.3 Star-catalogue and searching for matching stars

#### 3.3.1 Star Catalogue Generation

McBryde and Lightsey [64]

$$\mathbf{u} = \begin{bmatrix} \cos \alpha \cos \delta \\ \sin \alpha \cos \delta \\ \sin \delta \end{bmatrix} \quad (49)$$

$$m_i \leq m_{max} \quad (50)$$

$$m_j \leq m_{max} \quad (51)$$

$$\mathbf{u}_a^T \mathbf{u}_b \geq \cos \theta_{FOV} \quad (52)$$

#### 3.3.2 Candidate Matching

to be removed?

#### 3.3.3 Result Verification

to be removed?

#### 3.3.4 k-vector

The k-vector database is built a priori for some given working magnitude threshold and for the star tracker maximum angular aperture. Essentially, the k-vector table is a structural database of all catalogued star pairs that could possibly fit in the camera FOV over the whole sky. The star pairs are ordered with increasing inter-star angle. The data stored are the k index, the cosine of the interstar angle, and the master catalogue indices I[k] and J[k] of the kth star pair. The k-vector access logic is invoked in real time for a minimal set of star pairs in elementary measured star polygons (three for a triangle, six for a four-star pyramid, etc.); the fact that the vertices

between adjacent measured star pairs share a common catalogued star is the key observation leading to logic for efficiently identifying the stars by simply comparing the k-vector accessed catalogue indices from the several sets of candidate star pairs (which must contain the common measured pivot star, if it is in the catalog). Mortari and Rogers [21]

Mortari [19]

Mortari and Neta [20]

Trzeba dodać pogrubienia vectorów

$$z(x) = mx + q \quad (53)$$

$$m = \frac{y_{max} - y_{min} + \delta\epsilon}{n - 1} \quad (54)$$

$$q = y_{min} - m - \delta\epsilon \quad (55)$$

$$\epsilon \approx 22.2 \times 10^{-16} \quad (56)$$

$$\delta\epsilon = (n - 1)\epsilon \quad (57)$$

$$k(i) = j \quad \text{where} \quad s(j) \leq z(i) < s(j + 1) \quad (58)$$

or

$$k(i) = j \quad \text{where } j \text{ is the greatest index such } s(j) \leq y(I(i)) \text{ is satisfied.} \quad (59)$$

$$j_b = \left\lfloor \frac{y_a - q}{m} \right\rfloor \quad \text{and} \quad j_t = \left\lceil \frac{y_b - q}{m} \right\rceil \quad (60)$$

$$k_{start} = k(j_b) + 1 \quad \text{and} \quad k_{end} = k(j_t) \quad (61)$$

### 3.4 Attitude Determination

Jenssen et al. [68]

AIM (Attitude estimation using Image Matching) Delabie [24]

all Hall [31] Markley and Mortari [30]

### 3.4.1 q-method

$$\mathbf{s}_b = \mathbf{R}^{bi} \mathbf{s}_i \quad \mathbf{m}_b = \mathbf{R}^{bi} \mathbf{m}_i \quad (62)$$

$$\begin{aligned} J &= \frac{1}{2} \sum w_k (\mathbf{v}_{kb} - \mathbf{R}^{bi} \mathbf{v}_{ki})^T (\mathbf{v}_{kb} - \mathbf{R}^{bi} \mathbf{v}_{ki}) \\ &= \frac{1}{2} \sum w_k (\mathbf{v}_{kb}^T \mathbf{v}_{kb} + \mathbf{v}_{ki}^T \mathbf{v}_{ki} + 2 \mathbf{v}_{kb}^T \mathbf{R}^{bi} \mathbf{v}_{ki}) \end{aligned} \quad (63)$$

$$J = \sum w_k (1 - \mathbf{v}_{kb}^T \mathbf{R}^{bi} \mathbf{v}_{ki}) \quad (64)$$

$$g(\mathbf{R}) = \sum w_k \mathbf{v}_{kb}^T \mathbf{R}^{bi} \mathbf{v}_{ki} \quad (65)$$

$$\mathbf{R} = (q_4^2 - \mathbf{q}^T \mathbf{q}) \mathbf{I} + 2\mathbf{q}\mathbf{q}^T - 2q_4 \mathbf{q}^x \quad (66)$$

$$\bar{\mathbf{q}}^T \bar{\mathbf{q}} = 1 \quad (67)$$

$$g(\bar{\mathbf{q}}) = \bar{\mathbf{q}}^T \mathbf{K} \bar{\mathbf{q}} \quad (68)$$

$$\mathbf{K} = \begin{bmatrix} \mathbf{S} - \sigma \mathbf{I} & \mathbf{Z} \\ \mathbf{Z}^T & \sigma \end{bmatrix} \quad (69)$$

$$\mathbf{B} = \sum_{k=1}^N w_k (\mathbf{v}_{kb} \mathbf{v}_{ki}^T) \quad (70)$$

$$\mathbf{S} = \mathbf{B} + \mathbf{B}^T \quad (71)$$

$$\mathbf{Z} = \begin{bmatrix} B_{23} - B_{32} & B_{32} - B_{13} & B_{12} - B_{21} \end{bmatrix}^T \quad (72)$$

$$\sigma = \text{tr}[\mathbf{B}] \quad (73)$$

$$g'(\bar{\mathbf{q}}) = \bar{\mathbf{q}}^T \mathbf{K} \bar{\mathbf{q}} - \lambda \bar{\mathbf{q}}^T \bar{\mathbf{q}} \quad (74)$$

$$\mathbf{K} \bar{\mathbf{q}} = \lambda \bar{\mathbf{q}} \quad (75)$$

$$g(\bar{\mathbf{q}}) = \bar{\mathbf{q}}^T \mathbf{K} \bar{\mathbf{q}} = \bar{\mathbf{q}}^T \lambda \bar{\mathbf{q}} = \lambda \bar{\mathbf{q}}^T \bar{\mathbf{q}} = \lambda \quad (76)$$

### 3.4.2 Wahba's problem

The developed extended QUEST method described in this report builds upon the principles of Wahba's problem. The problem was first stated by Grace Wahba in 1965 [69]. Given two sets of vector observations, a rotation matrix  $\mathbf{M}$  can be found which minimizes the orientation error. This is an optimization problem, where the cost function is:

$$\sum_j^n \|\mathbf{r}_j - \mathbf{M} \mathbf{b}_j\| \quad (77)$$

For satellite attitude determination, the vectors  $\mathbf{r}_j$  for  $j \in \{1, n\}$  are the reference sensor data given in the NED frame. The vectors  $\mathbf{b}_j$  for  $j \in \{1, n\}$  are the measured sensor data in the BODY frame.  $\mathbf{M}$  is the least squares estimate of the rotation matrix which carries the known frame of reference into the satellite fixed frame of reference. The QUEST method uses this problem in order to minimize the attitude estimation error.

### 3.4.3 QUEST

improvement to quest implementation Cheng and Shuster [26]

kallman filtering Shuster [25]

$$\begin{aligned} J(\mathbf{q}) &= \frac{1}{2} \sum_{j=1}^n \frac{1}{\sigma_j^2} (\mathbf{b}_j - \mathbf{R}_b^i(\mathbf{q}) \mathbf{r}_j)^T (\mathbf{b}_j - \mathbf{R}_b^i(\mathbf{q}) \mathbf{r}_j) = \\ &\quad \frac{1}{2} \sum_{j=1}^n \frac{1}{\sigma_j^2} (\mathbf{b}_j^T \mathbf{b}_j - 2\mathbf{b}_j^T \mathbf{R}_b^i(\mathbf{q}) \mathbf{r}_j + \mathbf{r}_j^T \mathbf{r}_j) \end{aligned} \quad (78)$$

$$J(\mathbf{q}) = \sum_{j=1}^n \frac{1}{\sigma_j^2} (1 - \mathbf{b}_j^T \mathbf{R}_b^i(\mathbf{q}) \mathbf{r}_j) \quad (79)$$

### 3.4.4 TRIAD

a must

### 3.4.5 The Fast Optimal Attitude Matrix

to be removed?

### 3.4.6 DCM (Direction Cosine Matrix) - Singular Value Decomposition?

Juang et al. [29] and

McBryde and Lightsey [64]

$$\mathbf{B} = \sum_{i=1}^n \mathbf{b}_i \mathbf{r}_i^T \quad (80)$$

$$\mathbf{B} = \mathbf{U} \mathbf{S} \mathbf{V}^T \quad (81)$$

$$\mathbf{U}_+ = \mathbf{U} \begin{bmatrix} 1 & 0 & 0 \\ 0 & 1 & 0 \\ 0 & 0 & \det \mathbf{U} \end{bmatrix} \quad (82)$$

$$\mathbf{V}_+ = \mathbf{V} \begin{bmatrix} 1 & 0 & 0 \\ 0 & 1 & 0 \\ 0 & 0 & \det \mathbf{V} \end{bmatrix} \quad (83)$$

$$\mathbf{A} = \mathbf{U}_+ \mathbf{V}_+^T \quad (84)$$

## 4 Prototype

For now the following parts are finished in Python:

1. Centroiding
2. Planar Triangle Recognition with variations (nearly - without equation 44)
3. Pyramid alg ?
4. k-vector
5. QUEST (not started yet)

Testing

Kruijff et al. [70], Kruijff and vd Heiden [71], Dzamba and Enright [72],

## 5 Complete program

## 6 Testing of star-tracker

Tappe et al. [73]

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## List of Tables

1	Sensor Accuracy Ranges. Adapted from Hall [31] and Larson and Wertz [54] . . . . .	16
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## List of Figures

1	Example of Polish CubeSats . . . . .	12
2	CubeSat build - INSPIRE, Image [48] . . . . .	13
3	Earth's satellite navigation orbits [55] . . . . .	17
4	ECI frame, Image Larson and Wertz [54] . . . . .	19
5	BODY frame, Image Larson and Wertz [54] . . . . .	20
6	Startracker conceptual algorithm diagram . . . . .	24
7	Vector angle method, Image Gottlieb [66] . . . . .	28
8	Spherical Triangle Method, Image Cole and Crassidus [14] . . .	29
9	Planar Triangle Method, Image Cole and Crassidis [12] . . . .	30
10	Pyramid Method Flowchart, Image Mortari et al. [13] . . . . .	51

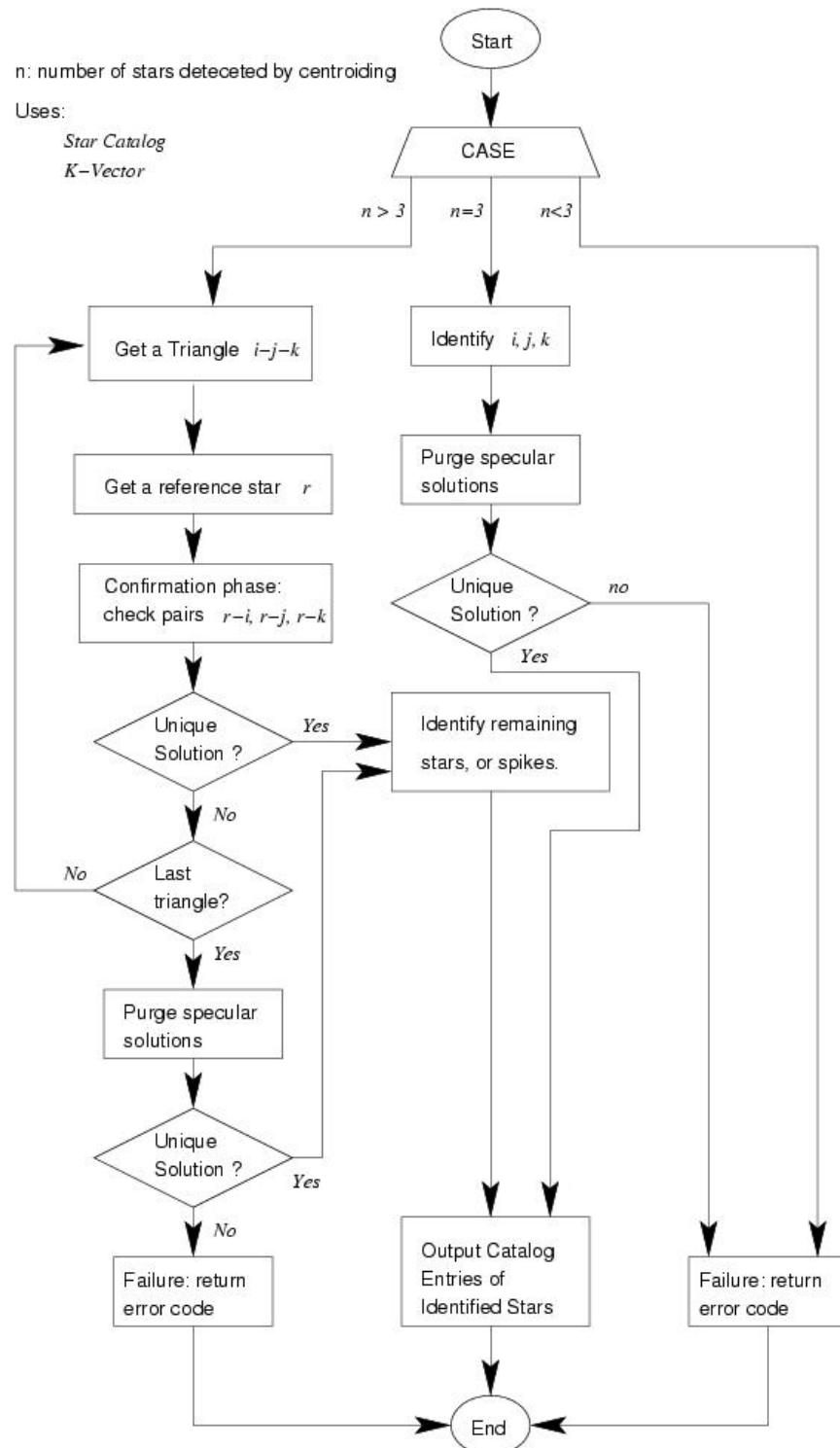


Figure 10: Pyramid Method Flowchart, Image Mortari et al. [13]