

## 5 Multiple degree-of-freedom systems

Until now we have only considered and modeled systems that can require one coordinate system to describe their motion. In this chapter we will develop the mathematical tools required to model multiple degree-of-freedom system that require multiple independent coordinates to describe their motion. As before, the equations that describe the motion of rigid bodies in space are developed from Newton's second law of motion. However, unlike before, there exists an independent equation for each body in motion. These equations are therefore coupled by the system and are often expressed in matrix notation such that the mass, damping, and stiffness matrices are easily defined.

### Review 5.1 Linear Algebra

Linear algebra allows for the efficient solving of these coupled equations.

The dot product allows us to multiply matrices and is defined as:

$$\begin{bmatrix} a & b \\ c & d \end{bmatrix} \begin{bmatrix} e \\ f \end{bmatrix} = \begin{bmatrix} ae + bf \\ ce + df \end{bmatrix} \quad (1)$$

Another arrangement of the same principle, in a format more related to vibrations, is:

$$\begin{bmatrix} a_1 + a_2 & b \\ c & d \end{bmatrix} \begin{bmatrix} e \\ f \end{bmatrix} = \begin{bmatrix} (a_1 + a_2)e + bf \\ ce + df \end{bmatrix} \quad (2)$$

The transpose of a matrix is an operator which flips a matrix over its diagonal. For a matrix  $A$ , the transpose  $A^T$  can be written as:

$$A = \begin{bmatrix} a & b \\ c & d \\ e & f \end{bmatrix} \rightarrow A^T = \begin{bmatrix} a & c & e \\ b & d & f \end{bmatrix} \quad (3)$$

A matrix is symmetric if  $A = A^T$ . Therefore, symmetric matrix must be square and can be written as:

$$A = \begin{bmatrix} a & b & c \\ d & e & f \\ g & h & i \end{bmatrix} = A^T = \begin{bmatrix} a & d & g \\ b & e & h \\ c & f & i \end{bmatrix}, \text{ where } b = d, c = g, f = h \quad (4)$$

The determinant of a matrix is a scalar value that is a function of the entries of a square matrix. The determinant characterizes the matrix and its linear map. The determinant is often written as  $\det(A)$ ,  $\det A$ , or  $|A|$ . For a  $2 \times 2$  matrix this is defined as:

$$\det(A) = ad - bc, \text{ when } A = \begin{bmatrix} a & b \\ c & d \end{bmatrix} \quad (5)$$

The inverse of a square matrix is such that  $AA^{-1} = A^{-1}A = I$  where  $I$  is the identity matrix:

$$I = \begin{bmatrix} 1 & 0 \\ 0 & 1 \end{bmatrix} \quad (6)$$

and the inverse of a  $2 \times 2$  matrix is defined as:

$$A^{-1} = \frac{1}{\det(A)} \begin{bmatrix} d & -b \\ -c & a \end{bmatrix}, \text{ when } A = \begin{bmatrix} a & b \\ c & d \end{bmatrix} \quad (7)$$

Matrices that do not have an inverse are called a singular matrix. **add notes on notation of vectors**

## 5.1 General Discussion on Mode Shapes

Studying and characterizing the natural frequencies of a system allows for the detailed investigation of the system response. Modern vibration analysis relies heavily on the concepts of mode shapes for various engineering tasks.

Mode shapes are not the displacement of a system, rather they describe the configurations into which a structure will naturally displace at a given frequency. For example, consider the 4-DOF system shown in figure 1 that represents a pole (i.e. cantilever beam). Assuming that the system experiences a linear response and using the mode-superposition method we can see that the displaced shape  $\vec{x}$  is a function of all of the mode shapes  $u_i$  and their corresponding participation factors  $q_i$ . Note that the mode shapes associated with the lower frequencies tend to provide the greatest contribution to structural response. As the frequencies that excite the modes increase, the mode shapes contribute less, are predicted less reliably, and are harder to measure. Therefore, the analysis of the system is often truncated after the first few modes and rarely exceeds the 10<sup>th</sup> mode.

Figure 1 shows a structure with  $N$  degrees of freedom that therefore has  $N$  corresponding mode shapes. Each mode shape is independent and normalized such that the maximum displacements are the same. The summation of the mode shapes multiplied by their corresponding participation factors ( $q_i$ ) yields the deflection of the structure.

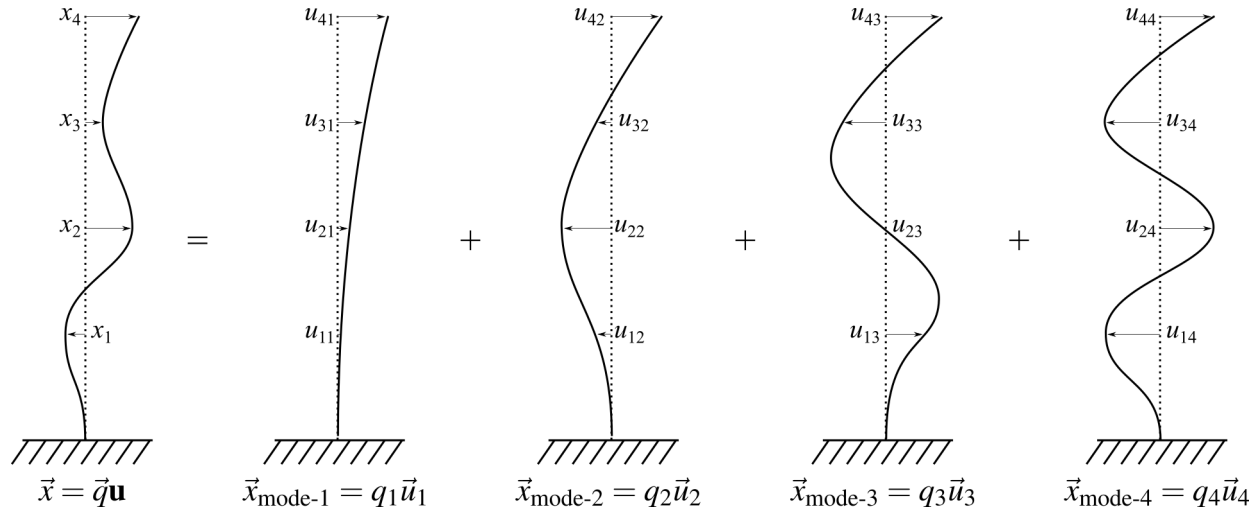


Figure 1: Deflection of a vertical cantilever,  $\vec{x}$ , is a function of the considered mode shapes  $u_i$  and their corresponding participation factors  $q_i$ .

## 5.2 Modeling Undamped Two Degree of Freedom Systems

Consider the undamped 2-DOF systems presented in figure 2. These system with a single mass capable of moving in two directions. To expand, figure 2(a) reports a mass that can move horizontally or vertically in space. However, this mass does not rotate during its movements. Moreover, figure 2(b) presents a system that rotates about the spring and displaces vertically. These are examples of 2 DOF systems because each system has two independent coordinate systems that express the movement of the mass.

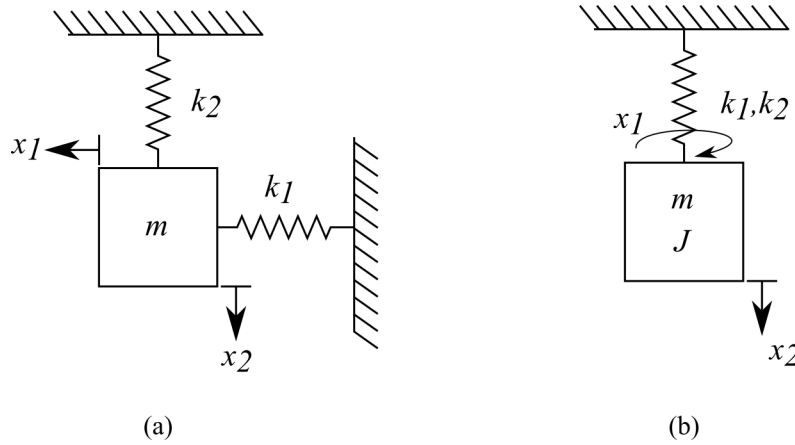


Figure 2: Examples of single mass 2-DOF systems that: (a) displace in the vertical and horizontal directions; and (b) rotates about the spring and displaces in the vertical direction.

Another example of a 2-DOF system with two masses, each with their own independent coordinate system, is presented in figure 3. The two coordinates that describe the systems movements are  $x_1$  and  $x_2$ .

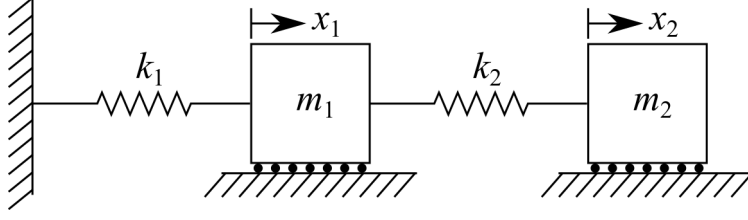


Figure 3: 2-DOF system with two masses and two independent confidante systems  $x_1$  and  $x_2$ .

### 5.2.1 Solution for the Two-Degree-of-Freedom System

Before we derive a model for undamped 2-DOF systems, let us first consider the solution to the system shown in figure 3. The solution consists of two equations, one for each mass. This solution will be derived in section 5.2.2 and is expressed by the coupled equations:

$$x_1(t) = A_1 \sin(\omega_1 t + \phi_1)u_{11} + A_2 \sin(\omega_2 t + \phi_2)u_{12} \quad (8)$$

$$x_2(t) = A_1 \sin(\omega_1 t + \phi_1)u_{21} + A_2 \sin(\omega_2 t + \phi_2)u_{22}, \quad \omega_1 \text{ or } \omega_2 \neq 0$$

These two equations can be written as a single equation in matrix form as:

$$\vec{x}(t) = A_1 \sin(\omega_1 t + \phi_1)\vec{u}_1 + A_2 \sin(\omega_2 t + \phi_2)\vec{u}_2, \quad \omega_1 \text{ or } \omega_2 \neq 0 \quad (9)$$

Where the bold text denotes vectors. Therefore, the vectors  $\vec{u}_1$  and  $\vec{u}_2$  are the mathematical expressions that “couple” or tie the equations together. Expanding these vectors shows:

$$\vec{x}(t) = \begin{bmatrix} x_1(t) \\ x_2(t) \end{bmatrix}, \quad \vec{u}_1 = \begin{bmatrix} u_{11} \\ u_{21} \end{bmatrix}, \quad \vec{u}_2 = \begin{bmatrix} u_{12} \\ u_{22} \end{bmatrix}, \quad (10)$$

The four key components of the solution expressed in equation 9 are:

1.  $\omega_1$  and  $\omega_2$  are the natural frequencies of the system. They are not the frequencies of the masses. The solution states that each of the masses oscillates at the two frequencies  $\omega_1$  and  $\omega_2$ . Moreover, consider the special case where the initial conditions are selected to force  $A_2 = 0$ , in this case, each mas would only oscillates at only one frequency,  $\omega_1$ .
2.  $A_1$  and  $A_2$  are the constants of integration and determine the amplitude of the system.
3.  $\phi_1$  and  $\phi_2$  represent the phase shift of the system
4.  $\vec{u}_1$  and  $\vec{u}_2$  are the first and second mode shapes of the system and couple the system together.

### 5.2.2 Deriving the Solution for the Two-Degree-of-Freedom System

To derive this solution for the system under consideration a FBD for figure 3 can be constructed for the forces acting on each mass. First we have to make the assumption that  $x_1 < x_2$ , this allows us to say that  $m_2$  pulls on  $m_1$  and results in:

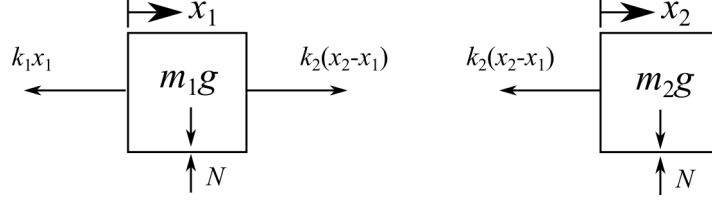


Figure 4: Free body diagram for the 2-DOF system presented in figure 3.

Applying Newton's second law and summing the forces on each mass in the horizontal direction yields:

$$\begin{aligned} m_1 \ddot{x}_1 &= -k_1 x_1 + k_2 (x_2 - x_1) \\ m_2 \ddot{x}_2 &= -k_2 (x_2 - x_1) \end{aligned} \quad (11)$$

These equations can be rearranged in terms of  $x_1$  and  $x_2$  as:

$$\begin{aligned} m_1 \ddot{x}_1 + (k_1 + k_2)x_1 - k_2 x_2 &= 0 \\ m_2 \ddot{x}_2 - k_2 x_1 + k_2 x_2 &= 0 \end{aligned} \quad (12)$$

where these are two coupled second-order differential equations that each require two initial conditions to solve. These initial conditions can be obtained from the displacement and velocity terms as:

$$\begin{aligned} x_1(0) &= x_{10} \\ \dot{x}_1(0) &= \dot{x}_{10} = v_{10} \\ x_2(0) &= x_{20} \\ \dot{x}_2(0) &= \dot{x}_{20} = v_{20} \end{aligned} \quad (13)$$

As before, these initial conditions will be the constants of integration used to solve the two second-order differential equations. This solution will provide the free response of each mass in the system. There is a multitude of ways to solve these two coupled second-order differential equations, however, here we will just consider a matrix notation solution. This matrix notation solution is used as this formulation is readably solved using computers and is expandable to more than 2 DOF.

To initiate the solution, let us first develop the matrix formulation of the two coupled ODEs:

$$\begin{bmatrix} m_1 & 0 \\ 0 & m_2 \end{bmatrix} \begin{bmatrix} \ddot{x}_1 \\ \ddot{x}_2 \end{bmatrix} + \begin{bmatrix} k_1 + k_2 & -k_2 \\ -k_2 & k_2 \end{bmatrix} \begin{bmatrix} x_1 \\ x_2 \end{bmatrix} = \begin{bmatrix} 0 \\ 0 \end{bmatrix} \quad (14)$$

This equation can also be expressed as the vector equation:

$$M\ddot{\vec{x}} + K\vec{x} = 0 \quad (15)$$

and is known as the EOM in vector form. In this formulation the mass matrix ( $M$ ) is defined as:

$$M = \begin{bmatrix} m_1 & 0 \\ 0 & m_2 \end{bmatrix} \quad (16)$$

while the stiffness matrix ( $K$ ) is:

$$K = \begin{bmatrix} k_1 + k_2 & -k_2 \\ -k_2 & k_2 \end{bmatrix} \quad (17)$$

along with the displacement, velocity, and acceleration matrices:

$$\vec{x} = \begin{bmatrix} x_1 \\ x_2 \end{bmatrix}, \quad \dot{\vec{x}} = \begin{bmatrix} \dot{x}_1 \\ \dot{x}_2 \end{bmatrix}, \quad \ddot{\vec{x}} = \begin{bmatrix} \ddot{x}_1 \\ \ddot{x}_2 \end{bmatrix} \quad (18)$$

Beyond these equations we can write the initial conditions as:

$$\vec{x}_0 = \begin{bmatrix} x_1(0) \\ x_2(0) \end{bmatrix}, \quad \dot{\vec{x}}_0 = \begin{bmatrix} \dot{x}_1(0) \\ \dot{x}_2(0) \end{bmatrix} \quad (19)$$

This simple connection between vibration analysis and matrix analysis allows computers to be used to solve large and complicated vibration problems quickly.

Recall that the 1-DOF version of the equation of motion was solved by calculating the values of the constants in an assumed harmonic solution. The same approach is applied here in order to solve for the displacement of the two-DOF system. This time, the the solution is assumed in the form:

$$\vec{x}(t) = \vec{u}e^{j\omega t} \quad (20)$$

where  $\vec{u}$  is a vector of constants to be determined and can be written as:

$$\vec{u} = \begin{bmatrix} u_1 \\ u_2 \end{bmatrix} \quad (21)$$

From before,  $\omega$  is also a constant to be determined. Again,  $j = \sqrt{-1}$ . In the same manner as before,  $e^{j\omega t}$  represents harmonic motion as  $e^{j\omega t} = \cos(\omega t) + j\sin(\omega t)$ . Taking the derivatives of  $\vec{x}(t) = \vec{u}e^{j\omega t}$  yields:

$$\dot{\vec{x}}(t) = j\omega\vec{u}e^{j\omega t} \quad (22)$$

$$\ddot{\vec{x}}(t) = -\omega^2\vec{u}e^{j\omega t} \quad (23)$$

Substituting this into the EOM in vector form ( $M\ddot{\vec{x}} + K\vec{x} = 0$ ) yields:

$$-\omega^2 M\vec{u}e^{j\omega t} + K\vec{u}e^{j\omega t} = 0 \quad (24)$$

or

$$(-\omega^2 M + K)\vec{u}e^{j\omega t} = 0 \quad (25)$$

As  $e^{j\omega t} \neq 0$  for any value of  $t$  and not allowing  $\vec{u}$  to be zero it can be determined that  $(-\omega^2 M + K)$  must satisfy the vector equation. Therefore,

$$(-\omega^2 M + K)\vec{u} = 0, \quad \vec{u} \neq 0 \quad (26)$$

This forms a homogeneous set of algebraic equations. To be useful, these equations have a nonzero solution for the system must exist. For this to be true, the the inverse of the coefficient matrix

$(-\omega^2 M + K)$  must not exist. To expand, assume that the the inverse of  $(-\omega^2 M + K)$  does exist, by multiplying both sides of the equation by  $(-\omega^2 M + K)^{-1}$  yields  $\vec{u} = 0$ . This is trivial solution (its not useful) as it no motion in the system is implied. Therefore, the logical connection can be drawn between the solution of equation and the inverse of the coefficient matrix  $(-\omega^2 M + K)$ .

Applying the singularity condition to the coefficient matrix of equation  $(-\omega^2 M + K)\vec{u} = 0$ ,  $\vec{u} \neq 0$  results a nonzero solution of  $\vec{u}$ . However, for this to exist the following must be true:

$$\det(-\omega^2 M + K) = 0 \quad (27)$$

Solving this expression results in one algebraic equation with one unknown ( $\omega$ ). Expanding the above equation to consider the values for the matrices  $M$  and  $K$  results in:

$$\det \begin{bmatrix} -\omega^2 m_1 + k_1 + k_2 & -k_2 \\ -k_2 & -\omega^2 m_2 + k_2 \end{bmatrix} = 0 \quad (28)$$

Using the definition of the determinant yields that the unknown quantity  $\omega^2$  must satisfy:

$$m_1 m_2 \omega^4 - (m_1 k_2 + m_2 k_1 + m_2 k_2) \omega^2 + k_1 k_2 = 0 \quad (29)$$

This expression is called the characteristic equation for the system and is used to determine the constants  $\omega_{1,2}$ , in the assumed form of the solution given by the assumed solution  $\vec{x}(t) = \vec{u}e^{j\omega t}$ , once the values of the physical parameters  $m_1$ ,  $m_2$ ,  $k_1$ , and  $k_2$  are known. Note that  $\omega_{1,2}$  are not in the characteristic equation, therefore, solving for  $\omega_{1,2}$  will be done by factoring the equation above to obtain two solutions  $\omega_1$  and  $\omega_2$ . The characteristic equation is in the form of the quadratic formula if you set  $x = \omega^2$ , as:

$$ax^2 + bx + c = 0 \quad (30)$$

After finding the value of  $\omega_{1,2}$  using the characteristic equation, the values in  $\vec{u}$  can be found using equation  $(-\omega^2 M + K)\vec{u} = 0$ ,  $\vec{u} \neq 0$  for each value of  $\omega^2$ . That is, for both  $\omega_1$  and  $\omega_2$  there is a vector  $\vec{u}$  that satisfies the equation. These solutions can be written as:

$$(-\omega_1^2 M + K)\vec{u}_1 = 0 \quad (31)$$

and

$$(-\omega_2^2 M + K)\vec{u}_2 = 0 \quad (32)$$

The direction of the vectors  $\vec{u}_1$  and  $\vec{u}_2$  can be obtained by solving the above expressions, however, the information regarding the magnitude of is not contained in this expression. To verify this, assume that  $\vec{u}_1$  satisfies the equation, therefore, the vector  $a\vec{u}_1$  also satisfies the equation where  $a$  is any nonzero number. Hence the vectors satisfying the above are of arbitrary magnitude.

The values obtained for  $\vec{u}_1$  and  $\vec{u}_2$  can now be combined with the assumed solution:

$$\vec{x}(t) = \vec{u}e^{j\omega t} \quad (33)$$

to form a set of solution:

$$\vec{x}(t) = \vec{u}_1 e^{-j\omega_1 t}, \quad \vec{u}_1 e^{j\omega_1 t}, \quad \vec{u}_2 e^{-j\omega_2 t}, \quad \vec{u}_2 e^{j\omega_2 t} \quad (34)$$

Since the equation to be solved is linear, the solution is the sum of these solutions. This results in:

$$\vec{x}(t) = (ae^{j\omega_1 t} + be^{-j\omega_1 t})\vec{u}_1 + (ce^{j\omega_2 t} + de^{-j\omega_2 t})\vec{u}_2 \quad (35)$$

where  $a$ ,  $b$ ,  $c$ , and  $d$  are the arbitrary constants of integration to be determined by the initial conditions. Applying Euler's formulas for the sin functions (where  $\omega_1$  or  $\omega_2 \neq 0$ ) reorganizes this equation as:

$$\vec{x}(t) = A_1 \sin(\omega_1 t + \phi_1)\vec{u}_1 + A_2 \sin(\omega_2 t + \phi_2)\vec{u}_2, \quad \omega_1 \text{ or } \omega_2 \neq 0 \quad (36)$$

Another way to write this equation is in the form:

$$\begin{bmatrix} x_1(t) \\ x_2(t) \end{bmatrix} = [\vec{u}_1 \quad \vec{u}_2] \begin{bmatrix} A_1 \sin(\omega_1 t + \phi_1) \\ A_2 \sin(\omega_2 t + \phi_2) \end{bmatrix}, \quad \omega_1 \text{ or } \omega_2 \neq 0 \quad (37)$$

Where the values for  $A_1$  and  $A_2$  can be obtained by setting applying the boundary conditions and taking the derivatives of the equations as done in the 1-DOF problems.

The final form of the equation provides a physical insight into the solution of the system. It states that the each mass in the system oscillates at both of the natural frequencies of the system ( $\omega_1$  and  $\omega_2$ ). Furthermore, the importance of the initial conditions can be understood. Assume that initial conditions are chosen that result in  $A_2 = 0$ , this cancels out the second natural frequency such that each mass oscillates at only one frequency,  $\omega_1$ . Moreover, the positions of the masses can be determined by the values of the vector  $\vec{u}_1$  at any given time. For this reason,  $\vec{u}_1$  is termed the first mode shape of the system. Likewise, if the opposite initial conditions are chosen such that  $A_1 = 0$ , then both system coordinates (e.g., masses in the systems we have studied) will oscillate at  $\omega_2$  and again, the positions can be obtained from the vector  $\vec{u}_2$ . Where  $\vec{u}_2$  is termed the second mode shape. The interactions between mode shapes and natural frequencies are very important and form the basis of several area in the field of vibrations.

**Example 5.1** Considering the following system:

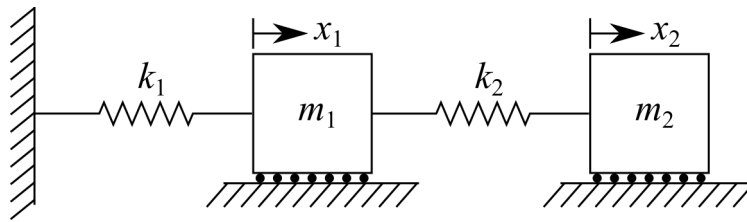


Figure 5: 2-DOF system with two masses and two independent confidante systems  $x_1$  and  $x_2$ .

Calculate response for the system if  $m_1=9$  kg,  $m_2=1$  kg,  $k_1 = 24$  N/m, and  $k_2 = 3$  N/m with the initial conditions  $x_{10} = 1$  mm,  $v_{10} = 0$  mm/s,  $x_{20} = 0$  mm, and  $v_{20} = 0$  mm/s.

**Solution:** We have already obtained a characteristic equation for this system. This is shown in Equation 29 and is given as:

$$m_1 m_2 \omega^4 - (m_1 k_2 + m_2 k_1 + m_2 k_2) \omega^2 + k_1 k_2 = 0 \quad (38)$$



Substituting our values into this obtains:

$$9 \cdot 1 \omega^4 - (9 \cdot 3 + 1 \cdot 24 + 1 \cdot 3) \omega^2 + 24 \cdot 3 = 0 \quad (39)$$

or

$$\omega^4 - 6\omega^2 + 8 = 0 \quad (40)$$

This can then be factored into:

$$(\omega^2 - 2)(\omega^2 - 4) = 0 \quad (41)$$

This results in solutions of  $\omega_1^2 = 2$  and  $\omega_2^2 = 4$ . Leading to:

$$\omega_1 = \pm\sqrt{2} \text{ rad/sec}, \quad \omega_2 = \pm 2 \text{ rad/sec} \quad (42)$$

Next, we need to obtain solutions for  $\vec{u}_1$  and  $\vec{u}_2$ . Having solved for  $\omega_1$  and  $\omega_2$  we can obtain. First, knowing  $\vec{u}_1 = [u_{11} u_{21}]^T$  and using  $\omega_1 = \sqrt{2}$  and the following equation:

$$(-\omega_1^2 M + K) \vec{u}_1 = 0 \quad (43)$$

yields simplified to

$$\left( -2 \begin{bmatrix} 9 & 0 \\ 0 & 1 \end{bmatrix} + \begin{bmatrix} 24+3 & -3 \\ -3 & 3 \end{bmatrix} \right) \begin{bmatrix} u_{11} \\ u_{21} \end{bmatrix} = \begin{bmatrix} 0 \\ 0 \end{bmatrix} \quad (44)$$

simplified to

$$\begin{bmatrix} 27-9 \cdot 2 & -3 \\ -3 & 3-2 \end{bmatrix} \begin{bmatrix} u_{11} \\ u_{21} \end{bmatrix} = \begin{bmatrix} 0 \\ 0 \end{bmatrix} \quad (45)$$

or

$$\begin{bmatrix} 9 & -3 \\ -3 & 1 \end{bmatrix} \begin{bmatrix} u_{11} \\ u_{21} \end{bmatrix} = \begin{bmatrix} 0 \\ 0 \end{bmatrix} \quad (46)$$

Taking the dot product of the matrix equation yields:

$$9u_{11} - 3u_{21} = 0, \text{ and } -3u_{11} + u_{21} = 0 \quad (47)$$

Both of these equations yield the same equation, that is:

$$\frac{u_{11}}{u_{21}} = \frac{1}{3} \quad (48)$$

As mentioned before, only the ratio of the elements is determined here. To show this is true it is easily seen that:

$$u_{11} = u_{21} \frac{1}{3} \rightarrow au_{11} = au_{21} \frac{1}{3} \quad (49)$$

To obtain a numerical value, we arbitrarily assign a value to one of the elements. Here, let  $u_{21} = 1$  so let  $u_{11} = 1/3$ . Therefore,

$$\vec{u}_1 = \begin{bmatrix} \frac{1}{3} \\ 1 \end{bmatrix} \quad (50)$$

The same processes can be used for obtaining  $\vec{u}_1$  using  $\omega_2 = 2$ , this results in:

$$\begin{bmatrix} -9 & -3 \\ -3 & -1 \end{bmatrix} \begin{bmatrix} u_{12} \\ u_{22} \end{bmatrix} = \begin{bmatrix} 0 \\ 0 \end{bmatrix} \quad (51)$$

Taking the dot product of the matrix equation yields:

$$-9u_{12} - 3u_{22} = 0, \text{ and } -3u_{12} - u_{22} = 0 \quad (52)$$

Both of these equations yield the same equation, that is:

$$\frac{u_{12}}{u_{22}} = -\frac{1}{3} \quad (53)$$

Again, assuming  $u_{22} = 1$  this can be rearranged into  $\vec{u}_2$  as:

$$\vec{u}_2 = \begin{bmatrix} -\frac{1}{3} \\ 1 \end{bmatrix} \quad (54)$$

Where  $\vec{u}_1$  and  $\vec{u}_2$  represent only the directions and shape of the mode shapes and not the magnitude of the mode shapes. Now that we have the mode shapes, we can solve for the initial conditions  $A_1$  and  $A_2$ . To do this, let us use the following formulation of the solution:

$$\begin{bmatrix} x_1(t) \\ x_2(t) \end{bmatrix} = [\vec{u}_1 \quad \vec{u}_2] \begin{bmatrix} A_1 \sin(\omega_1 t + \phi_1) \\ A_2 \sin(\omega_2 t + \phi_2) \end{bmatrix}, \quad \omega_1 \text{ or } \omega_2 \neq 0 \quad (55)$$

Adding our values for the problem at  $t = 0$  this becomes:

$$\begin{bmatrix} 1 \\ 0 \end{bmatrix} = \begin{bmatrix} \frac{1}{3} & -\frac{1}{3} \\ 1 & 1 \end{bmatrix} \begin{bmatrix} A_1 \sin(\phi_1) \\ A_2 \sin(\phi_2) \end{bmatrix} \quad (56)$$

and after applying the dot product:

$$\begin{bmatrix} 1 \\ 0 \end{bmatrix} = \begin{bmatrix} \frac{1}{3}A_1 \sin(\phi_1) - \frac{1}{3}A_2 \sin(\phi_2) \\ A_1 \sin(\phi_1) + A_2 \sin(\phi_2) \end{bmatrix} \quad (57)$$

Next we can differentiate the equation for  $x(t)$  to obtain the velocity solution. Adding our values for the problem at  $t = 0$  obtains:

$$\begin{bmatrix} \dot{x}_1(0) \\ \dot{x}_2(0) \end{bmatrix} = \begin{bmatrix} v_{10} \\ v_{20} \end{bmatrix} = \begin{bmatrix} 0 \\ 0 \end{bmatrix} = \begin{bmatrix} \frac{\sqrt{2}}{3}A_1 \cos(\phi_1) - \frac{2}{3}A_2 \cos(\phi_2) \\ \sqrt{2}A_1 \cos(\phi_1) + 2A_2 \cos(\phi_2) \end{bmatrix} \quad (58)$$

Now that we have 4 equations for 4 unknowns we can use these equations to solve for  $A_1$ ,  $A_2$ ,  $\phi_1$ , and  $\phi_2$ . The 4 equations are:

$$3 = A_1 \sin(\phi_1) - A_2 \sin(\phi_2) \quad (59)$$

$$0 = A_1 \sin(\phi_1) + A_2 \sin(\phi_2) \quad (60)$$

$$0 = \sqrt{2}A_1 \cos(\phi_1) - 2A_2 \cos(\phi_2) \quad (61)$$

$$0 = \sqrt{2}A_1 \cos(\phi_1) + 2A_2 \cos(\phi_2) \quad (62)$$

Setting these last two equations equal to each other yields:

$$0 = \sqrt{2}A_1 \cos(\phi_1) + 2A_2 \cos(\phi_2) = \sqrt{2}A_1 \cos(\phi_1) - 2A_2 \cos(\phi_2) \quad (63)$$

or:

$$0 = -4A_2 \cos(\phi_2) \quad (64)$$

For this equation to be true,  $\phi_2 = \frac{\pi}{2}$ . Therefore, applying this to  $0 = \sqrt{2}A_1 \cos(\phi_1) + 2A_2 \cos(\phi_2)$  results in:

$$0 = \sqrt{2}A_1 \cos(\phi_1) \quad (65)$$

where again, for this equation to be true,  $\phi_1 = \frac{\pi}{2}$ . Now the first two equations become:

$$3 = A_1 - A_2 \quad (66)$$

$$0 = A_1 + A_2 \quad (67)$$

Where this shows us that  $A_1 = \frac{3}{2}$  and  $A_2 = -\frac{3}{2}$ . Therefore, now that we have the initial conditions we can find a solution for the temporal response of each mass. Using the equations from before:

$$x_1(t) = A_1 \sin(\omega_1 t + \phi_1)u_{11} + A_2 \sin(\omega_2 t + \phi_2)u_{12} \quad (68)$$

$$x_2(t) = A_1 \sin(\omega_1 t + \phi_1)u_{21} + A_2 \sin(\omega_2 t + \phi_2)u_{22} \quad (69)$$

And applying our obtained values

$$x_1(t) = \frac{3}{2} \sin(\sqrt{2}t + \frac{\pi}{2})\frac{1}{3} + \left(-\frac{3}{2}\right) \sin(2t + \frac{\pi}{2})\left(-\frac{1}{3}\right) \quad (70)$$

$$x_2(t) = \frac{3}{2} \sin(\sqrt{2}t + \frac{\pi}{2}) + \left(-\frac{3}{2}\right) \sin(2t + \frac{\pi}{2}) \quad (71)$$

results in:

$$x_1(t) = \frac{1}{2} \left( \sin(\sqrt{2}t + \frac{\pi}{2}) + \sin(2t + \frac{\pi}{2}) \right) \quad (72)$$

$$x_2(t) = \frac{3}{2} \left( \sin(\sqrt{2}t + \frac{\pi}{2}) - \sin(2t + \frac{\pi}{2}) \right) \quad (73)$$

These results can be plotted as:

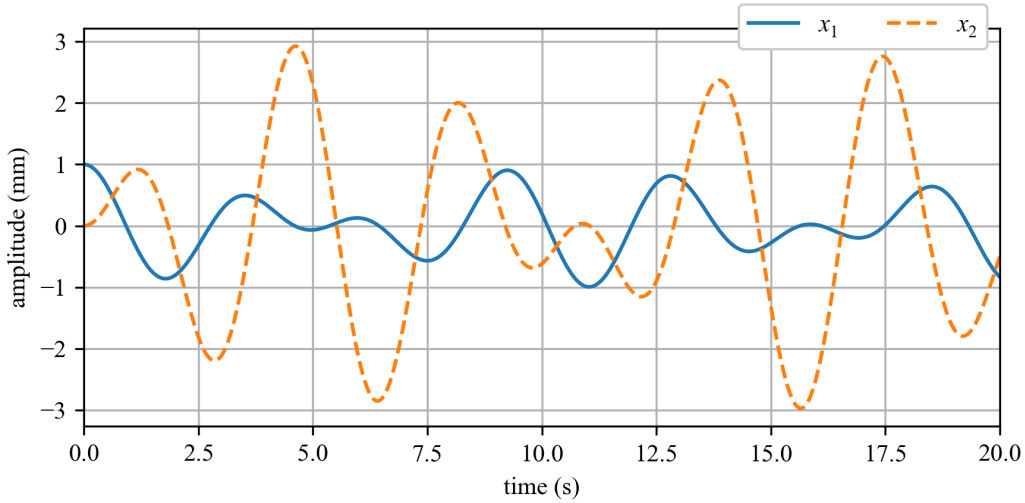


Figure 6: Temporal response for each of the rigid bodies in the 2-DOF system.

**Example 5.2** Mode shapes can be better understood through a graphical representation. To do this, consider the 2-DOF system presented in figure 7(a). Assuming that  $x_1 < x_2$  the FBD for the system is expressed in figure 7(b).

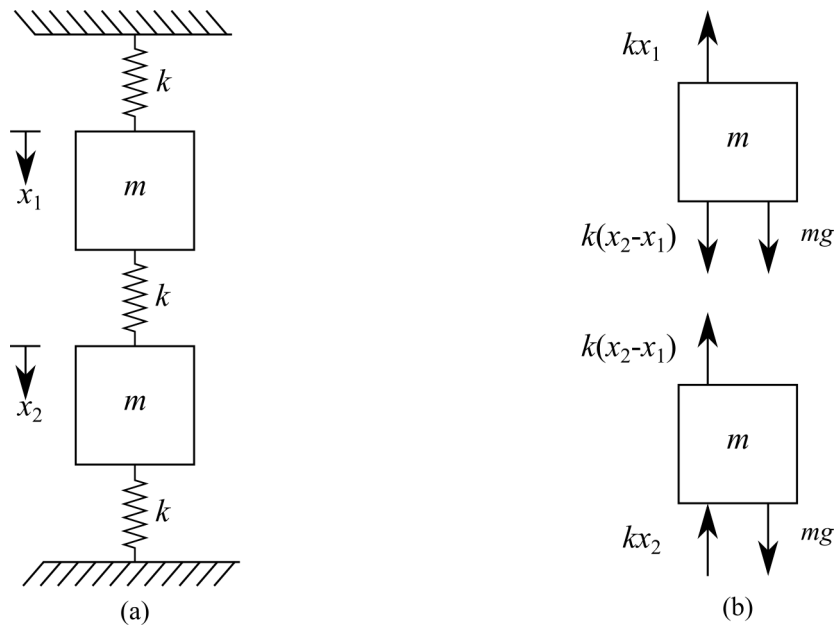


Figure 7: (a) 2-DOF system with two masses arranged in a vertical configuration; and (b) FBD of system.

For simplicity, all masses and spring stiffness are considered equal and that  $m = 1$  and  $k = 1$ . From the previous investigations in this text we know that the forces caused by gravity will

cancel out. Therefore, the EOM for the system can be written as:

$$\begin{aligned} m\ddot{x}_1 &= -kx_1 + k(x_2 - x_1) \\ m\ddot{x}_2 &= -k(x_2 - x_1) - kx_2 \end{aligned} \quad (74)$$

These equations can be written in matrix notation as:

$$\begin{bmatrix} m & 0 \\ 0 & m \end{bmatrix} \begin{bmatrix} \ddot{x}_1 \\ \ddot{x}_2 \end{bmatrix} + \begin{bmatrix} 2k & -k \\ -k & 2k \end{bmatrix} \begin{bmatrix} x_1 \\ x_2 \end{bmatrix} = \begin{bmatrix} 0 \\ 0 \end{bmatrix} \quad (75)$$

Substituting the values of the matrices  $M$  and  $K$  into this expression  $\det(-\omega^2 M + K) = 0$  yields:

$$\det \begin{bmatrix} -\omega^2 m + 2k & -k \\ -k & -\omega^2 m + 2k \end{bmatrix} = 0 \quad (76)$$

The determinant yields that the unknown quantity,  $\omega^2$ , must satisfy:

$$m^2 \omega^4 - 4km\omega^2 + 3k = 0 \quad (77)$$

Therefore,

$$\omega_1 \pm \sqrt{\frac{k}{m}} = 1 \text{ rad/sec}, \quad \omega_2 \pm \sqrt{\frac{3k}{m}} = \sqrt{3} \text{ rad/sec} \quad (78)$$

Now, we need to obtain solutions for  $\vec{u}_1$  and  $\vec{u}_2$ . Knowing  $(-\omega_1^2 M + K)\vec{u}_1 = 0$  yields:

$$\begin{bmatrix} 1 & -1 \\ -1 & 1 \end{bmatrix} \begin{bmatrix} u_{11} \\ u_{21} \end{bmatrix} = \begin{bmatrix} 0 \\ 0 \end{bmatrix} \quad (79)$$

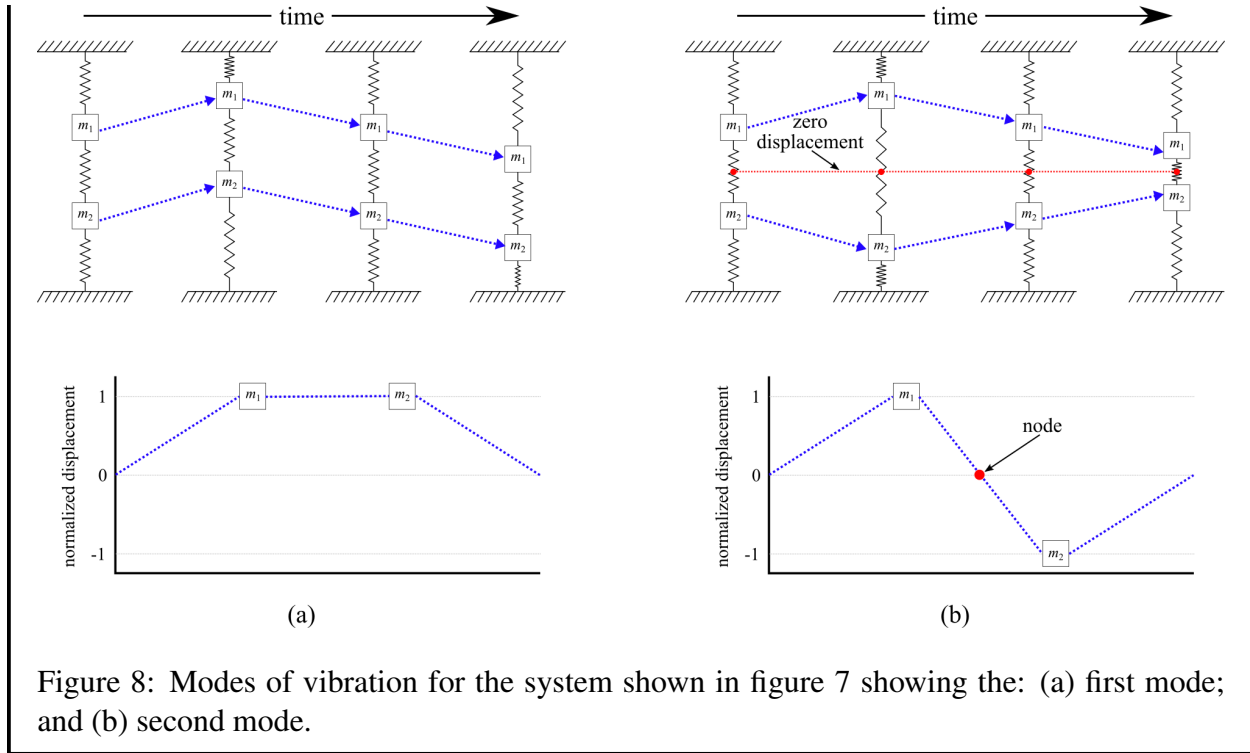
Taking the dot product of the matrix equation yields:

$$u_{11} - u_{21} = 0, \text{ and } -u_{11} + u_{21} = 0 \quad (80)$$

Setting  $u_{11} = 1$  results in  $u_{21} = 1$ . The same processes can be performed for  $\vec{u}_2$  to show that if we set  $u_{12} = 1$ ,  $u_{22} = -1$ . Therefore, the mode shapes can be expressed as:

$$[\vec{u}_1 \quad \vec{u}_2] = \begin{bmatrix} u_{11} & u_{12} \\ u_{21} & u_{22} \end{bmatrix} = \begin{bmatrix} 1 & 1 \\ 1 & -1 \end{bmatrix} \quad (81)$$

The displacement of the masses as a function of time and the general mode shape plots are graphically represented in figure 8. In the 2-DOF system considered here, the second mode shape has a spot at the center of the middle spring that does not move (i.e. has zero displacement). This point is called a node. Nodes correspond to points in the mode shape where the displacement is always zero. Furthermore, the displacement of the node points remain zero at all times, as diagrammed in the top-right of figure 8.



### 5.3 Eigenvalue-based Solution for Natural Frequencies and Mode Shapes

The process of calculating the mode shapes presented in section 5.2 is long and tedious. Therefore, methods that can be easily deployed on computers are of great interest to the practitioner. An eigenvalue-based solution that takes advantage of the symmetry in the  $M$  and  $K$  matrices and can be easily implemented on a computer is discussed in this section.

#### Review 5.2 Eigenvalues and Eigenvectors

In linear algebra, eigenvalues and eigenvectors are concepts that appear prominently in the analysis of linear transformations. By definition, If  $\mathbf{v}$  is a vector (in vector space  $V$  over a field  $F$ ) and  $T$  is a linear transformation into itself, then  $\mathbf{v}$  is an eigenvector of  $T$  if  $T(\mathbf{v})$  is a scalar multiple of  $\mathbf{v}$ :

$$T(\mathbf{v}) = \lambda \mathbf{v} \quad (82)$$

where  $\lambda$  is a scalar in the field  $F$ , known as the eigenvalue associated with the eigenvector  $\mathbf{v}$ . If the linear transformation is expressed in the form of an  $n \times n$  matrix  $A$ , then the eigenvalue equation for a linear transformation above can be rewritten as the matrix multiplication

$$A\mathbf{v} = \lambda \mathbf{v} \quad (83)$$

where  $\mathbf{v}$  is a  $n \times 1$  matrix of the eigenvectors. For the matrix  $A$ , eigenvalues and eigenvectors can be used to decompose the matrix.

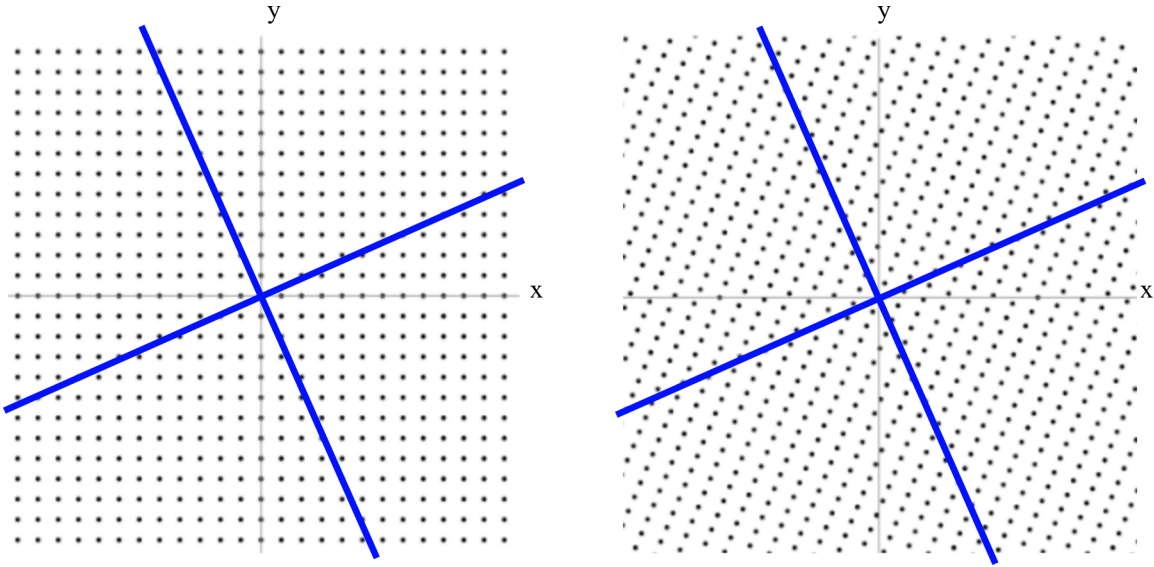


Figure 9: The stretching and shearing of the plane on which a matrix lies where the eigenvectors of the matrix are denoted by the blue lines and are the special directions such that point on the lines will just slide on them.

Need to add a discussion on generalized eigenvalues, normal, orthogonal, and orthonormal matrices.

A matrix is positive definite if the scalar  $\mathbf{x}^T A \mathbf{x}$  is positive for any non-zero vector  $\mathbf{x}$  comprised of real numbers:

$$\mathbf{x}^T A \mathbf{x} > 0 \quad (84)$$

The Cholesky decomposition of a real positive-definite matrix  $A$  is a decomposition of the form:

$$A = LL^T \quad (85)$$

where  $L$  is the lower triangular matrix of  $A$ .

### Review 5.3 Cholesky Decomposition

The vast majority of mass and stiffness matrices are symmetric and positive definite due to the physical meaning of these matrices. Therefore,  $M$  can be factored into two terms using the Cholesky decomposition:

$$M = LL^T \quad (86)$$

For diagonal mass matrices (all the mass values lie along the diagonal of the matrix) the Cholesky decomposition ( $L$ ) is defined as:

$$L = M^{1/2} = \begin{bmatrix} \sqrt{m_1} & 0 \\ 0 & \sqrt{m_2} \end{bmatrix} \quad (87)$$

This expression factors into:

$$M = M^{1/2} M^{1/2} \quad (88)$$

Moreover, the inverse of the diagonal matrix ( $M^{1/2}$ ) is denoted as  $M^{-1/2}$  and defined as:

$$L^{-1} = M^{-1/2} = \begin{bmatrix} \frac{1}{\sqrt{m_1}} & 0 \\ 0 & \frac{1}{\sqrt{m_2}} \end{bmatrix} \quad (89)$$

### 5.3.1 Deriving the Eigenvalue-based Solution

To derive an eigenvalue-based solution for calculating the natural frequencies and mode shapes let us consider the previously derived EOM for a undamped 2-DOF system:

$$M\ddot{\vec{x}} + K\vec{x} = 0 \quad (90)$$

This expression can be transformed into a symmetric eigenvalue problem, allowing us to leverage the strengths of symmetric eigenvalue mathematics and computer solvers. To perform this transform, we set  $\vec{x} = M^{-1/2}\vec{q}$  and multiply the equation by  $M^{-1/2}$  such that the EOM becomes:

$$M^{-1/2}MM^{-1/2}\ddot{\vec{q}} + M^{-1/2}KM^{-1/2}\vec{q} = 0 \quad (91)$$

As  $M^{-1/2}MM^{-1/2}$  is equal to the identity matrix  $I$  and defining  $M^{-1/2}KM^{-1/2}$  as the mass normalized stiffness  $\tilde{K}$  yields the simplified expression:

$$I\ddot{\vec{q}} + \tilde{K}\vec{q} = 0 \quad (92)$$

where  $\tilde{K} = M^{-1/2}KM^{-1/2}$  is analogous to  $\sqrt{k/m}$  from the 1-DOF system.

As before, a solution is found by assuming a solution, taking the derivatives of the solution and substituting it into the EOM. Following these steps and assuming a solution of:

$$\vec{q} = \mathbf{v}e^{j\omega t} \quad (93)$$

results in an EOM in the form:

$$-\mathbf{v}\omega^2 e^{j\omega t} + \tilde{K}\mathbf{v}e^{j\omega t} = 0 \quad (94)$$

driving out the nonzero scalar  $e^{j\omega t}$  and rearranging the above expression results in:

$$\tilde{K}\mathbf{v} = \omega^2\mathbf{v} \quad (95)$$

Knowing that  $\mathbf{v} \neq 0$ , as a vector with zeros would mean no motion is present in the system, this equation can be expressed in a typical eigenvalue formulation:

$$\tilde{K}\mathbf{v} = \lambda\mathbf{v} \quad (96)$$

where  $\lambda = \omega^2$ . Or more importantly,  $\omega = \sqrt{\lambda}$ . As  $\tilde{K}$  is symmetric, this is a symmetric eigenvalue problem. Moreover, the vector  $\lambda$  represent the eigenvalues of the system. Given that we set  $\vec{x} = M^{-1/2}\vec{q}$ , the eigenvectors are not a direct representation of the mode shapes. To develop a link between the eigenvectors and mode shapes we first need to normalize the lengths of the



eigenvectors obtained by solving equation 96 to that of a unit vector. The norm of a unit vector is defined as:

$$1 = ||\mathbf{v}|| = \sqrt{\mathbf{v}^T \mathbf{v}} = \sqrt{\sum_{i=1}^n (v_i^2)} \quad (97)$$

where a scalar is used in conjunction with vector such that  $\alpha \mathbf{u} = 1$  computed the normalized unit vector directions as outlined in example 5.3. In general, a nonzero vector of any length can be normalized by calculating:

$$\frac{1}{\sqrt{\mathbf{v}^T \mathbf{v}}} \mathbf{v} \quad (98)$$

We can relate the eigenvectors to the modes shapes by a factor of the mass matrix:

$$\vec{u}_1 = M^{-1/2} \vec{v}_1 \quad (99)$$

The important thing to remember is that the natural frequencies are the square root of the eigenvalues and the mode shapes are related to the eigenvectors through the mass matrix.

**Example 5.3** Normalize the vector  $\vec{v}_1 = [1/3 \ 1]^T$

**Solution:** To normalize the vector  $\vec{v}_1$ , a scalar ( $\alpha$ ) is calculated to make  $\alpha \mathbf{v} = 1$ . Therefore, following the definition of an orthogonal vector:

$$(\alpha \vec{v}_1)^T (\alpha \vec{v}_1) = 1 \quad (100)$$

or:

$$\alpha [1/3 \ 1] \alpha \begin{bmatrix} 1/3 \\ 1 \end{bmatrix} = \alpha^2 (1/9 + 1) = 1 \quad (101)$$

Therefore,  $\alpha = 3/\sqrt{10}$ . Resulting in a normalized unit vector of  $\alpha \vec{v}_1 = [1/\sqrt{10} \ 3/\sqrt{10}]^T$

**Example 5.4** Consider the system presented in example 5.1 and repeated below where  $m_1=9$  kg,  $m_2=1$  kg,  $k_1 = 24$  N/m, and  $k_2 = 3$  N/m with the initial conditions  $x_{10} = 1$  mm,  $v_{10} = 0$  mm/s,  $x_{20} = 0$  mm, and  $v_{20} = 0$  mm/s. Calculate the natural frequencies and the mode shapes using the eigenvalue solution.

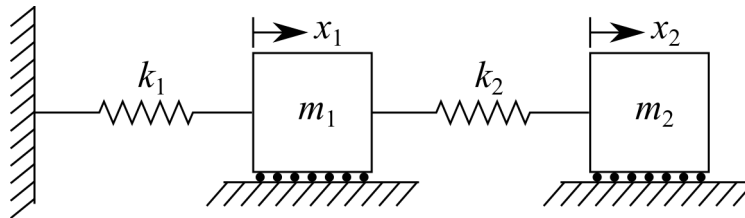


Figure 10: 2-DOF system with two masses and two independent confidante systems  $x_1$  and  $x_2$ .

**Solution:** Writing the mass and stiffness matrix of the system as:

$$M = \begin{bmatrix} 9 & 0 \\ 0 & 1 \end{bmatrix} \quad (102)$$

and

$$K = \begin{bmatrix} 27 & -3 \\ -3 & 3 \end{bmatrix} \quad (103)$$

we can compute  $\tilde{K}$  using the following expression:

$$\tilde{K} = M^{-1/2} K M^{-1/2} \quad (104)$$

where  $K M^{-1/2}$  is computed first to maintain symmetry. This results in:

$$K M^{-1/2} = \begin{bmatrix} 27 & -3 \\ -3 & 3 \end{bmatrix} \begin{bmatrix} \frac{1}{3} & 0 \\ 0 & 1 \end{bmatrix} = \begin{bmatrix} 9 & -3 \\ -1 & 3 \end{bmatrix} \quad (105)$$

and:

$$\tilde{K} = M^{-1/2} K M^{-1/2} = \begin{bmatrix} \frac{1}{3} & 0 \\ 0 & 1 \end{bmatrix} \begin{bmatrix} 9 & -3 \\ -1 & 3 \end{bmatrix} \begin{bmatrix} \frac{1}{3} & 0 \\ 0 & 1 \end{bmatrix} = \begin{bmatrix} 3 & -1 \\ -1 & 3 \end{bmatrix} \quad (106)$$

Now a solution must be obtained for the eigenvalue problem:

$$\tilde{K} \mathbf{v} = \lambda \mathbf{v} \quad (107)$$

While this can be obtained using computers for such a simple case it is more appropriate to solve this expression by hand. Therefore, the above expression can be rewritten as:

$$(\tilde{K} - \lambda I) \mathbf{v} = 0 \quad (108)$$

as  $\mathbf{v} \neq 0$  the matrix must be singular, resulting in the determinant of the matrix equaling zero. Or:

$$\det \begin{bmatrix} 3 - \lambda & -1 \\ -1 & 3 - \lambda \end{bmatrix} = 0 \quad (109)$$

This can be expanded to the characteristic equation:

$$\lambda^2 - 6\lambda + 8 = 0 \quad (110)$$

with the roots (eigenvalues):

$$\lambda_1 = 2 \text{ and } \lambda_2 = 4 \quad (111)$$

Therefore,  $\omega_1 = \sqrt{2}$  and  $\omega_2 = 2$ . These are the same values computed in example 5.1. The eigenvectors for  $\lambda_1$  are computed as:

$$(\tilde{K} - \lambda_1 I) \mathbf{v} = 0 \quad (112)$$

or:

$$\begin{bmatrix} 3 - 2 & -1 \\ -1 & 3 - 2 \end{bmatrix} \begin{bmatrix} v_{11} \\ v_{21} \end{bmatrix} = \begin{bmatrix} 0 \\ 0 \end{bmatrix} \quad (113)$$

This results in two dependent scalar equations:

$$v_{11} - v_{21} = 0 \text{ and } -v_{11} + v_{21} = 0 \quad (114)$$

That show us that  $v_{11} = v_{21}$  or  $\vec{v}_1 = [1 \ 1]^T$ . Therefore, using  $(\alpha\vec{v}_1)^T(\alpha\vec{v}_1) = 1$  we obtain:

$$\alpha[1 \ 1]\alpha \begin{bmatrix} 1 \\ 1 \end{bmatrix} = \alpha^2(2) = 1 \quad (115)$$

or  $\alpha = 1/\sqrt{2}$ . This allows us to normalize the vector knowing  $\alpha\vec{v}_1 = 1$ , resulting in a normalized vector of:

$$\alpha\vec{v}_1 = \frac{1}{\sqrt{2}} \begin{bmatrix} 1 \\ 1 \end{bmatrix} \quad (116)$$

A similar processes is followed for  $\lambda_2 = 4$  that leads to the normalized vector

$$\alpha\vec{v}_2 = \frac{1}{\sqrt{2}} \begin{bmatrix} -1 \\ 1 \end{bmatrix} \quad (117)$$

Lastly, the normalized eigenvectors can be converted to mode shapes using  $\mathbf{u} = M^{-1/2}\mathbf{v}$ . Resulting in:

$$\vec{u}_1 = \begin{bmatrix} \frac{1}{3} & 0 \\ 0 & 1 \end{bmatrix} \begin{bmatrix} 1 \\ 1 \end{bmatrix} = \begin{bmatrix} \frac{1}{3} \\ 1 \end{bmatrix} \quad (118)$$

and:

$$\vec{u}_2 = \begin{bmatrix} \frac{1}{3} & 0 \\ 0 & 1 \end{bmatrix} \begin{bmatrix} -1 \\ 1 \end{bmatrix} = \begin{bmatrix} -\frac{1}{3} \\ 1 \end{bmatrix} \quad (119)$$

Therefore, the first mode shape is  $[1 \ 1]$  while the second mode shape is  $[\frac{1}{3} \ -\frac{1}{3}]$ . Again, this is using our prior definition of mode shapes to set  $[1 \ 1]$  as the first mode shape. Note that these are the same mode shape vectors as computed in example 5.1.

Expanding on equation 99, the one can go from the eigenvector to the mode shapes through:

$$\vec{u}_1 = M^{-1/2}\vec{v}_1 \text{ and } \vec{v}_1 = M^{1/2}\vec{u}_1 \quad (120)$$

therefore, it can be seen that the eigenvectors and mode shapes are related through the term  $\vec{v}_1$ .

## 5.4 Transfer-function method

As in 1-DOF systems, transfer functions can be used to solve for the temporal response of 2-DOF systems under a variety of inputs. Again, the transfer function of a differential equation is defined as the ratio of the Laplace transform of the output (system response) to the Laplace transform of the input (forcing function). Moreover, the procedure for using the Laplace transform to solve the equations of motion is the same and follows three steps:

1. Take the Laplace transform of both sides of the EOM while treating the time derivatives symbolically.
2. Solve for  $X(s)$  in the obtained equation.
3. Apply the inverse transform  $x(t) = \mathcal{L}[X(s)]^{-1}$

### Example 5.5 2-DOF System Subjected to Impulse

Two masses are connected through a spring, as shown in figure 11.

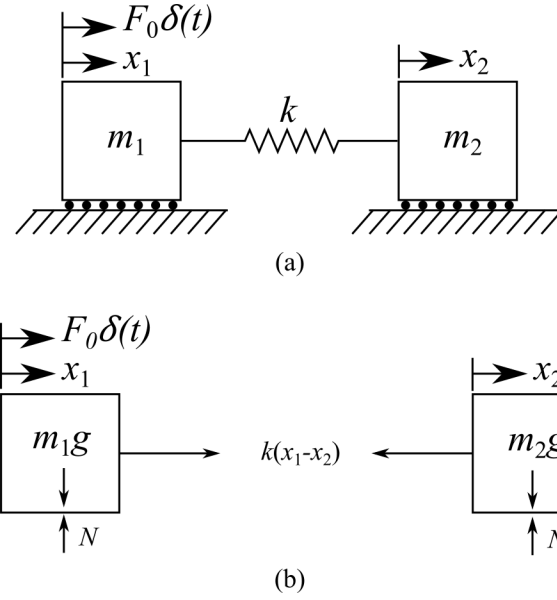


Figure 11: 2-DOF system subjected to an impulse showing: (a) system, and (b) FBD.

Assuming that  $x_1$  displaces more than  $x_2$ , the equations of motion are:

$$\begin{aligned} m_1 \ddot{x}_1 + k(x_1 - x_2) &= F_0 \delta(t) \\ m_1 \ddot{x}_1 + k(x_2 - x_1) &= 0 \end{aligned} \quad (121)$$

Taking the Laplace of both equations (step 1) yields:

$$\begin{aligned} (m_1 s^2 + k)X_1(s) - kX_2(s) &= F_0 \\ -kX_1(s) + (m_2 s^2 + k)X_2(s) &= 0 \end{aligned} \quad (122)$$

solving these two equations for  $X_1$  and  $X_2$  (step 2) results in:

$$\begin{aligned} X_1(s) &= \frac{F_0(m_2 s^2 + k)}{s^2[m_1 m_2 s^2 + k(m_1 + m_2)]} \\ X_2(s) &= \frac{F_0 k}{s^2[m_1 m_2 s^2 + k(m_1 + m_2)]} \end{aligned} \quad (123)$$

Using partial fractions, or a symbolic toolbox in MATLAB or Python, these expressions can be rewritten as:

$$X_1(s) = \frac{F_0}{m_1 + m_2} \left( \frac{1}{s^2} + \frac{m_2}{\omega m_1} \frac{\omega}{s^2 + \omega^2} \right) \quad (124)$$

$$X_2(s) = \frac{F_0}{m_1 + m_2} \left( \frac{1}{s^2} + \frac{1}{\omega} \frac{\omega}{s^2 + \omega^2} \right)$$

where:

$$\omega^2 = k \left( \frac{1}{m_1} + \frac{1}{m_2} \right) \quad (125)$$

Taking the inverse transform of the expressions for  $X_1(s)$  and  $X_2(s)$  (step 3) results in expressions in the time domain and yields:

$$x_1(t) = \frac{F_0}{m_1 + m_2} \left( t + \frac{m_2}{\omega m_1} \sin(\omega t) \right) \quad (126)$$

$$x_2(t) = \frac{F_0}{m_1 + m_2} \left( t + \frac{1}{\omega} \sin(\omega t) \right)$$

Considering a system where  $F_0 = 10$  N,  $m_1 = 1000$  kg,  $m_2 = 1000$  kg, and  $k = 1500$  N/m the temporal response is annotated in figure 12.

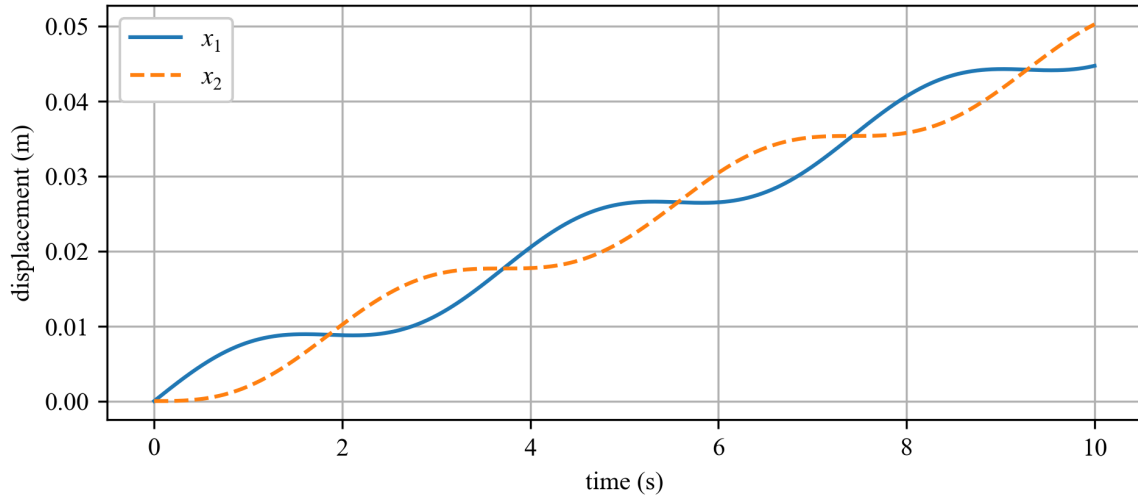


Figure 12: Temporal response for the considered 2-DOF system subjected to a impact load.

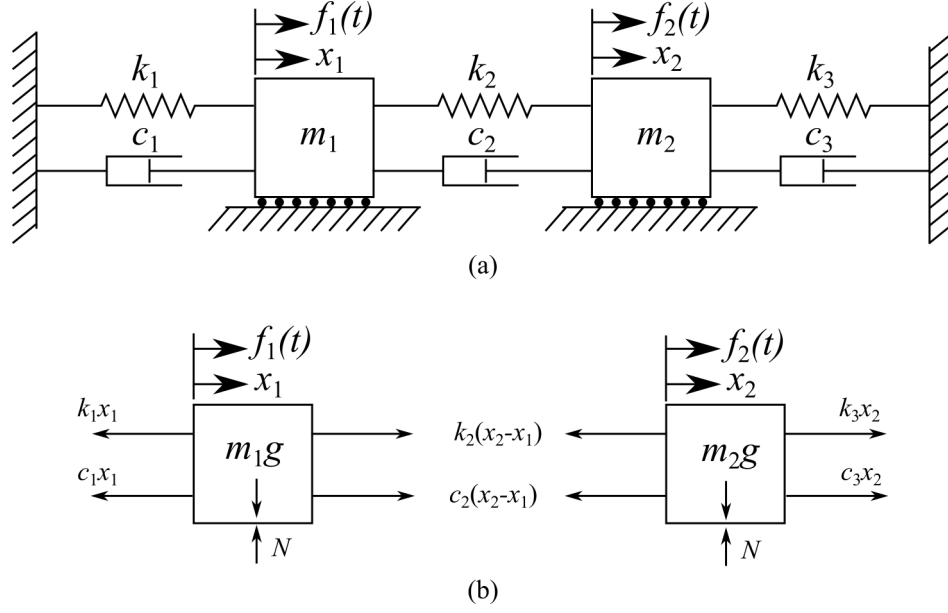


Figure 13: Forced 2-DOF damped system showing: (a) system, and (b) FBD.

## 5.5 Cramer's Rule for Solving 2-DOF Systems

Cramer's rule is an explicit formula for the solution of a system of linear equations with as many equations as unknown. Cramer's rule is valid whenever the system has a unique solution and can be used as a more generalized approach to solve for the temporal solution to a 2-DOF. Consider the 2-DOF systems shown in figure 13, where the two coupled equations of motion are expressed as:

$$\begin{aligned} m_1\ddot{x}_1 + (c_1 + c_2)\dot{x}_1 - c_2\dot{x}_2 + (k_1 + k_2)x_1 - k_2x_2 &= 0 \\ m_2\ddot{x}_2 + (c_2 + c_3)\dot{x}_2 - c_2\dot{x}_1 + (k_2 + k_3)x_2 - k_2x_1 &= 0 \end{aligned} \quad (127)$$

As before, taking the Laplace of the EOM (while ignoring the initial conditions) changes the equation from the temporal domain to the complex  $s$ -plane. This yields:

$$\begin{aligned} m_1s^2X_1(s) + (c_1 + c_2)sX_1(s) - c_2sX_2(s) + (k_1 + k_2)X_1(s) - k_2X_2(s) &= F_1(s) \\ m_2s^2X_2(s) + (c_2 + c_3)sX_2(s) - c_2sX_1(s) + (k_2 + k_3)X_2(s) - k_2X_1(s) &= F_2(s) \end{aligned} \quad (128)$$

these equations can be rearranged in terms of  $X_1$  and  $X_2$  as follows:

$$\begin{aligned} [m_1s^2 + (c_1 + c_2)s + (k_1 + k_2)]X_1(s) - [c_2s + k_2]X_2(s) &= F_1(s) \\ [m_2s^2 + (c_2 + c_3)s + (k_2 + k_3)]X_2(s) - [c_2s + k_2]X_1(s) &= F_2(s) \end{aligned} \quad (129)$$

These equations show two linear equations in terms of  $X_1$  and  $X_2$  that can be solved for using Cramer's rule, resulting in the expression:

$$\begin{aligned} X_1(s) &= \frac{D_1(s)}{D(s)} \\ X_2(s) &= \frac{D_2(s)}{D(s)} \end{aligned} \quad (130)$$

where:

$$D_1 = \begin{vmatrix} F_1(s) & -(c_2s + k_2) \\ F_2(s) & m_2s^2 + (c_2 + c_3)s + (k_2 + k_3) \end{vmatrix} \quad (131)$$

$$= [m_2s^2X_2(s) + (c_2 + c_3)s + (k_2 + k_3)]F_1(s) + (c_2s + k_2)F_2(s)$$

$$D_2 = \begin{vmatrix} m_1s^2 + (c_1 + c_2)s + (k_1 + k_2) & F_1(s) \\ -(c_2s + k_2) & F_2(s) \end{vmatrix} \quad (132)$$

$$= [m_1s^2 + (c_1 + c_2)s + (k_1 + k_2)]F_2(s) + (c_2s + k_2)F_1(s)$$

$$D = \begin{vmatrix} m_1s^2 + (c_1 + c_2)s + (k_1 + k_2) & m_2s^2 + (c_2 + c_3)s + (k_2 + k_3) \\ -(c_2s + k_2) & -(c_2s + k_2) \end{vmatrix} \quad (133)$$

$$= m_1m_2s^4 + [m_2(c_1 + c_3) + m_1(c_2 + c_3)]s^3$$

$$+ [m_2(k_1 + k_2) + m_1(k_2 + k_3) + c_1c_2 + c_2c_3 + c_3c_1]s^2$$

$$+ [(k_1 + k_2)(c_2 + c_3) + c_1k_2 + c_1k_3 - c_2k_2 + c_2k_3]s$$

$$+ (k_1k_2 + k_2k_3 + k_3k_1)$$

The denominator,  $D(s)$  is a 4<sup>th</sup> polynomial in  $s$  and is the characteristic polynomial of the system. The system is considered a 4<sup>th</sup> order system because the characteristic polynomial of the system is of order 4.

**Example 5.6** content... content... content...

example 5.11 in Rao ed 6 has a good example of this nature.

## 5.6 Frequency Response Transfer Functions

Rao ed 5.11, p 526

## 5.7 Computational Methods for 2-DOF Systems

### 5.7.1 MATLAB

#### Example 5.7 Eigenvalue Problem

Determine the natural frequencies and mode shapes of the system shown in figure 3, where  $m_1=9$  kg,  $m_2=1$  kg,  $k_1 = 24$  N/m, and  $k_2 = 3$  N/m with the initial conditions  $x_{10} = 1$  mm,  $v_{10} = 0$  mm/s,  $x_{20} = 0$  mm, and  $v_{20} = 0$  mm/s with the EOM expressed as: **this needs more text**

$$\begin{bmatrix} m_1 & 0 \\ 0 & m_2 \end{bmatrix} \begin{bmatrix} \ddot{x}_1 \\ \ddot{x}_2 \end{bmatrix} + \begin{bmatrix} k_1 + k_2 & -k_2 \\ -k_2 & k_2 \end{bmatrix} \begin{bmatrix} x_1 \\ x_2 \end{bmatrix} = \begin{bmatrix} 0 \\ 0 \end{bmatrix} \quad (134)$$

Using the eigenvalue method, the problem can be solved for as:

### Listing 1: Sample code from Matlab

```
clear
close
clc

% define the M and K matrix
M = [9 0; 0 1]
K = [24+3 -3; -3 3]

% build the M inverse squareroot and mass normalized stiffness
  matrix
M_inv_sqr = sqrt(inv(M))
K_mass_norm = M_inv_sqr*K*M_inv_sqr

% Using , K_mass_norm*v=lambda*v
[v,lambda] = eig(K_mass_norm)

% Solve for natural frequencies
omega_1 = sqrt(lambda(1,1))
omega_2 = sqrt(lambda(2,2))

% solve for the mode shapes
u_1 = M_inv_sqr*v(:,1)
u_2 = M_inv_sqr*v(:,2)
```

where  $\omega_1=1.41$  rad/sec and  $\omega_2=2$  rad/sec while  $\vec{u}_1 = [-0.2357 \ -0.7071]$  and  $\vec{u}_2 = [-0.2357 \ 0.7071]$ .

### Example 5.8 Roots of Quadratic Equation

#### Listing 2: Sample code from Matlab

```
clear
close
clc

% for the function f(x) = x^4 -8x +12 =0
roots([1 0 0 -8 12])
```

### Example 5.9 Roots of Quadratic Equation

#### Listing 3: Function for Matlab code

```
function f = matlab_function_dfunc(t,y)
```



```

%UNTITLED Summary of this function goes here
% Detailed explanation goes here
f=zeros(4,1)
f(1) = y(2)
f(2) = cos(3*t) - 4*y(2) + y(4) - 5*y(1) + 2*y(3)
f(3) = y(2)
f(4) = cos(3*t) + 0.5*y(2) - y(4) + y(1) - 1.5*y(3)
end

```

Listing 4: Sample code from Matlab

```

clear
close
clc

% define the time stamps of the system
tspan = [0: 0.1 : 20]

% define the initial conditions
y0 = [0.2; 1.0; 0.0; 8.0];

% solve the 2-DOF system using a ODE solver
[t,y] = ode23('matlab_function_dfunc',tspan, y0)

% plot the results
figure()
subplot(211)
plot(t,y(:,1))
xlabel('time (s)')
ylabel('x1(t) (m)')
subplot(212)
plot(t,y(:,3))
xlabel('time (s)')
ylabel('x1(t) (m)')

```

### 5.7.2 Python

## 5.8 Multiple Degrees of Freedom

This chapter introduces methodologies for the solving of systems with more than 2-DOF. As shown in Chapter 5, 2-DOF systems can be solved analytically using 2 EOM coupled through their mode shapes. However, these methods become tedious when extended to systems with damping or even beyond beyond 2-DOF system.

### Example 5.10 Multiple Mode Shapes

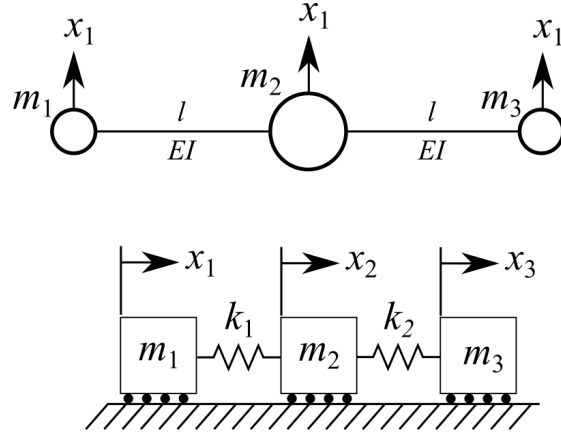


Figure 14: A Beechcraft Baron in flight<sup>1</sup> along with the Free-Free 3-DOF model simplified as a mass-spring model. **This footnote is messed up.**

Modeling the vibrations of a dual engine airplane as a three-degree-of-freedom system can be done as shown in figure 14 where the fuselage is a center mass, and the engines are point masses suspended by cantilevers from the center mass. The stiffness of the wing corresponds to the modulus of the wing  $E$  and its moment of inertia  $I$ . assuming that  $m_1 = m_3 = 1m$ ,  $m_2 = 4m$ , and  $k = \frac{3EI}{l}$ , the EOM can be written as:

$$m \begin{bmatrix} 1 & 0 & 0 \\ 0 & 2 & 0 \\ 0 & 0 & 1 \end{bmatrix} \begin{bmatrix} \ddot{x}_1 \\ \ddot{x}_2 \\ \ddot{x}_3 \end{bmatrix} + \frac{3EI}{l} \begin{bmatrix} 3 & -3 & 0 \\ -3 & 6 & -3 \\ 0 & -3 & 3 \end{bmatrix} \begin{bmatrix} x_1 \\ x_2 \\ x_3 \end{bmatrix} = \begin{bmatrix} 0 \\ 0 \\ 0 \end{bmatrix} \quad (135)$$

calculate the natural frequencies and mode shapes of the system and plot the mode-shapes in relation to the considered Beechcraft Baron.

#### Solution using the mass normalized stiffness matrix $\tilde{K}$ .

Solving for the modes shapes using the mass normalized stiffness matrix  $\tilde{K}$  requires solving for  $M^{-1/2}$  and  $\tilde{K}$  such that:

$$M^{-1/2} = \frac{1}{\sqrt{m}} \begin{bmatrix} 1 & 0 & 0 \\ 0 & 0.5 & 0 \\ 0 & 0 & 1 \end{bmatrix} \quad (136)$$

$$\tilde{K} = M^{-1/2} K M^{-1/2} = \frac{3EI}{l} \begin{bmatrix} 3 & -1.5 & 0 \\ -1.5 & 1.5 & -1.5 \\ 0 & -1.5 & 3 \end{bmatrix} \quad (137)$$

Than, the eigenvalue problem, formulated as:

$$\tilde{K} \mathbf{v} = \lambda \mathbf{v} \quad (138)$$

is solved for the eigenvalues and eigenvectors, resulting in:

$$\lambda_1 = 0, \lambda_2 = 3, \lambda_3 = 4.5 \quad (139)$$

$$\vec{v}_1 = \begin{bmatrix} 0.408248 \\ 0.816497 \\ 0.408248 \end{bmatrix}, \vec{v}_2 = \begin{bmatrix} -0.707107 \\ 0.0 \\ -0.707107 \end{bmatrix}, \vec{v}_3 = \begin{bmatrix} 0.57735 \\ -0.57735 \\ 0.57735 \end{bmatrix} \quad (140)$$

When the eigenvalue problem is solved using the mass normalized stiffness matrix  $\tilde{K}$  the natural frequencies are  $\omega_i = \sqrt{\lambda_i}$  while the mode shapes are derived from the eigenvectors as  $\vec{u} = M^{-1/2}\vec{v}$ . This results in:

$$\omega_1 = 0 \text{ rad/sec}, \omega_2 = 1.732 \text{ rad/sec}, \omega_3 = 2.121 \text{ rad/sec} \quad (141)$$

$$\vec{u}_1 = \begin{bmatrix} 0.408248 \\ 0.408248 \\ 0.408248 \end{bmatrix}, \vec{u}_2 = \begin{bmatrix} -0.707107 \\ 0.0 \\ -0.707107 \end{bmatrix}, \vec{u}_3 = \begin{bmatrix} 0.57735 \\ -0.288675 \\ 0.57735 \end{bmatrix} \quad (142)$$

Next, normalizing the mode shapes by the norm of the vector as  $\vec{u}_i = \frac{\vec{u}_i}{|\vec{u}_i|}$  results in:

$$\vec{u}_1 = \begin{bmatrix} 0.57735 \\ 0.57735 \\ 0.57735 \end{bmatrix}, \vec{u}_2 = \begin{bmatrix} -0.707107 \\ 0.0 \\ 0.707107 \end{bmatrix}, \vec{u}_3 = \begin{bmatrix} 0.666667 \\ -0.333333 \\ 0.666667 \end{bmatrix} \quad (143)$$

these mode-shapes can then be plotted as:

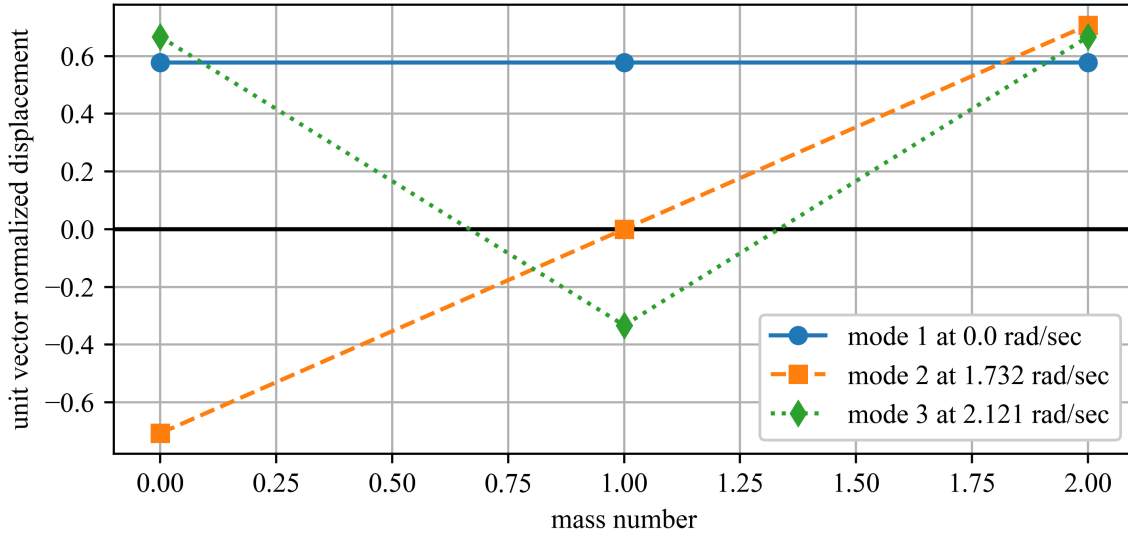


Figure 15: The unit vector normalized displacement of the mode shapes solved for using the mass normalized stiffness matrix  $\tilde{K}$ .

### Solution using the generalized eigenvalue approach.

The mode shapes can also be solved for using the generalized eigenvalue approach where the eigenvalue problem is written as:

$$\lambda M \mathbf{v} = K \mathbf{v} \quad (144)$$

solving for the eigenvalues and eigenvectors yields:

$$\lambda_1 = 0, \lambda_2 = 3, \lambda_3 = 4.5 \quad (145)$$

$$v_1 = \begin{bmatrix} -0.57735 \\ -0.57735 \\ -0.57735 \end{bmatrix}, v_2 = \begin{bmatrix} 0.707107 \\ 0.0 \\ -0.707107 \end{bmatrix}, v_3 = \begin{bmatrix} 0.666667 \\ -0.333333 \\ 0.666667 \end{bmatrix} \quad (146)$$

Note that the eigenvalues are the same as those solved for using the normalized stiffness matrix approach while the eigenvectors appear to be different. This is because for generalized eigenvalue approach  $u_i = v_i \alpha$  where  $\alpha$  is a scaling value. Note that here  $\alpha = 1$  as the mode shapes are already normalized. Note, that some software tools may not return normalized modes and the nodes should be normalized by the norm of the vector ( $u_i = \frac{u_i}{|u_i|}$ ) to achieve the same mode shapes as before. Moreover, recall that the information stored in the eigenvectors are just the direction of the transform. Therefore, the mode shapes are correct but some amplitudes are reversed but as the mode shapes only report the shape these are corrected. Plotting the mode shapes results in:

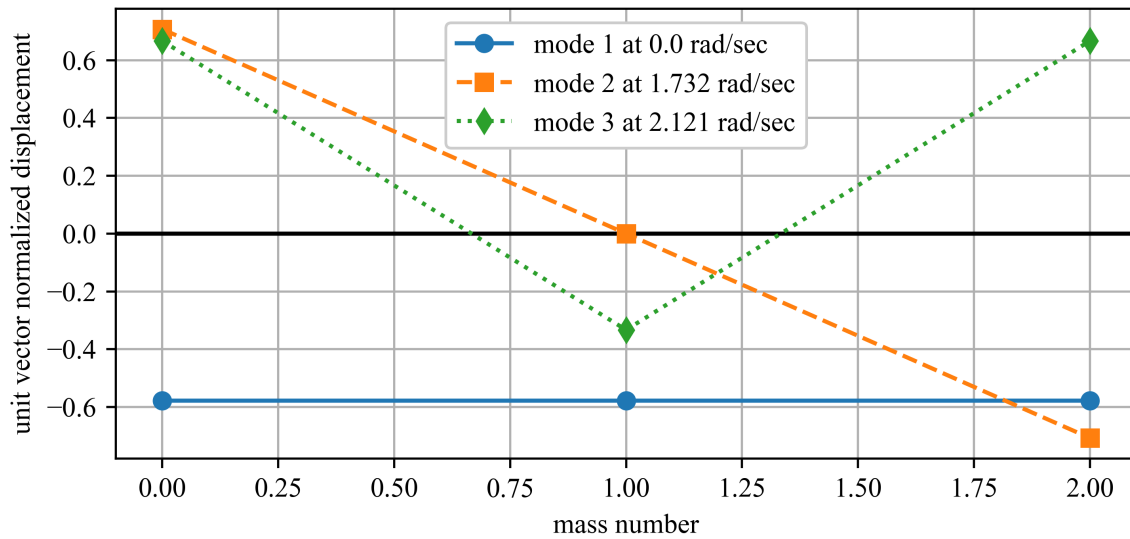


Figure 16: The unit vector normalized displacement of the mode shapes solved for using the generalized eigenvalue approach.

<sup>a</sup>“A Beechcraft Baron 58 in flight” by San Diego Air & Space Museum Archives, Public Domain.

## 5.9 Modal Analysis

Modal analysis is the study of a system's dynamic properties and is done in the frequency domain. Consider a system with  $n$  degrees of motion, modal analysis allows for the uncoupling of the EOM into  $n$  single-degree-of-freedom system (represented as 2<sup>nd</sup>-order DOF systems) where the displacements of the masses are expressed as the linear summations of the normal modes of the system. If every mode shape is considered, the solution is equivalent to the solution obtained from the original  $n^{\text{th}}$ -degree-of-freedom system.

Consider the generic multidegree-of-freedom system under external forces, expressed as:

$$M\ddot{\vec{x}} + K\vec{x} = \vec{F} \quad (147)$$

where damping is not considered and the vector  $\vec{F}$  is a set of deterministic inputs. To expand this equation by modal analysis, the eigenvalue problem must first be solved. The generalized eigenvalue problem is written at:

$$\lambda M\mathbf{v} = K\mathbf{v} \quad (148)$$

For the  $n^{\text{th}}$ -degree-of-freedom, the generalized eigenvalue problem can be simplified to:

$$\omega_i^2 M\vec{v}_i = K\vec{v}_i \quad (149)$$

Considering that the total displacement of the system, expressed as  $\vec{x}(t)$ , is the summation of the displacement of each of the noncontributing modes; assuming a linear system, the temporal response of the system can be written as:

$$\vec{x}(t) = q_1(t)\vec{v}_1 + q_2(t)\vec{v}_2 + q_3(t)\vec{v}_3 + \cdots + q_n(t)\vec{v}_n \quad (150)$$

where the time-dependent generalized scalars  $q_1(t), q_1(t), \dots, q_1(t)$  are the modal participation coefficients (also called principal coordinates). Defining the modal matrix  $P$  as:

$$P = [\vec{v}_1 \ \vec{v}_2 \ \vec{v}_3 \ \cdots \ \vec{v}_n] \quad (151)$$

Therefore, the linear combination of the normal modes (equation 150) can be more concisely written as:

$$\vec{x}(t) = P\vec{q}(t) \quad (152)$$

where  $\vec{q}(t) = [q_1 \ q_2 \ q_3 \ \cdots \ q_n]^T$ . Next, the relationship that relates the physical space to the modal space for the acceleration component is written as:

$$\ddot{\vec{x}}(t) = P\ddot{\vec{q}}(t) \quad (153)$$

combining these two terms results in the EOM that can be written as:

$$MP\ddot{\vec{q}}(t) + KP\vec{q}(t) = \vec{F} \quad (154)$$

To convert the EOM into the standard form, first the  $P^T$  is multiplied through the equation as:

$$P^T MP\ddot{\vec{q}}(t) + P^T KP\vec{q}(t) = P^T \vec{F} \quad (155)$$

If the modes are normalized, the following is true:

$$P^T M P = I \quad (156)$$

where  $I$  is the identity matrix and

$$P^T K P = \begin{bmatrix} \nearrow & 0 & 0 \\ 0 & \omega^2 & 0 \\ 0 & 0 & \searrow \end{bmatrix} \quad (157)$$

Moreover, defining  $\vec{Q}(t) = P^T \vec{F}$  results in a EOM expressed as:

$$\ddot{\vec{q}}(t) + \begin{bmatrix} \nearrow & 0 & 0 \\ 0 & \omega^2 & 0 \\ 0 & 0 & \searrow \end{bmatrix} \vec{q}(t) = \vec{Q}(t) \quad (158)$$

For a system with  $n$  degrees of freedom, this equation can be broken down into:

$$\ddot{q}_i(t) + \omega_i^2 q_i(t) = \vec{Q}(t), \quad i = 1, 2, \dots, n \quad (159)$$

This expression is the same ODE that we have solved multiple times in this text. Therefore, we know the solution to be:

$$q_i(t) = q_{i0} \cos(\omega_i t) + \frac{\dot{q}_{i0}}{\omega_i} \sin(\omega_i t) \quad (160)$$

Lastly, to solve for a solution in the modal space, the initial conditions that were given in the physical space must be converted to the modal space. This can be done by generalizing the velocities in terms of the modal matrix:

$$\vec{q}(0) = P^T M \vec{x}(0) \quad (161)$$

$$\dot{\vec{q}}(0) = P^T M \vec{v}(0) \quad (162)$$

### Example 5.11 Free vibration response

Solve for the free vibration response of the 2-DOF presented in figure 17 using modal analysis. Show the temporal response for the entire system for its first 20 seconds using the full modal reconstruction and the reconstruction truncated to just include the first mode. Also, plot the variations in the modal participation coefficients though time. Apply the parameters,  $f_1 = 0$  kg,  $f_2 = 0$  N,  $m_1 = 10$  kg,  $m_2 = 1$  kg,  $k_1 = 30$  N/m,  $k_2 = 5$  N/m,  $k_3 = 1$  N/m,  $x_1(0) = 1$  mm,  $x_2(0) = 0$  mm,  $v_1(0) = 0$  mm, and  $v_2(0) = 0$  mm.

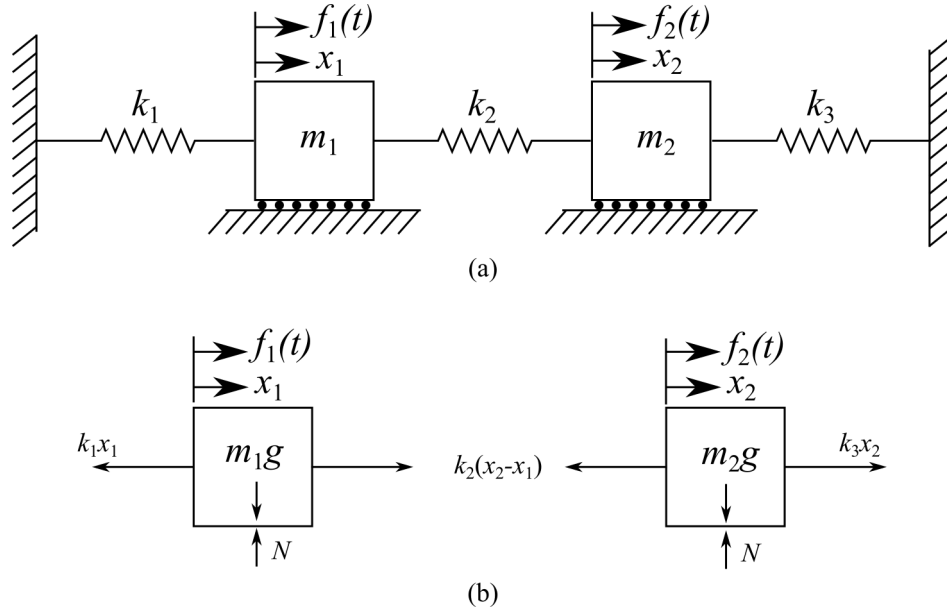


Figure 17: Forced 2-DOF damped system showing: (a) system, and (b) FBD.

The equations of motion that couple the system are:

$$\begin{aligned} m_1 \ddot{x}_1 + (k_1 + k_2)x_1 - k_2 x_2 &= 0 \\ m_2 \ddot{x}_2 + (k_2 + k_3)x_2 - k_2 x_1 &= 0 \end{aligned} \quad (163)$$

In matrix form, these become:

$$\begin{bmatrix} m_1 & 0 \\ 0 & m_2 \end{bmatrix} \begin{bmatrix} \ddot{x}_1 \\ \ddot{x}_2 \end{bmatrix} + \begin{bmatrix} k_1 + k_2 & -k_2 \\ -k_2 & k_2 + k_3 \end{bmatrix} \begin{bmatrix} x_1 \\ x_2 \end{bmatrix} = \begin{bmatrix} 0 \\ 0 \end{bmatrix} \quad (164)$$

$$\vec{x}(0) = \begin{bmatrix} 1 \\ 0 \end{bmatrix}, \quad \vec{v}(0) = \begin{bmatrix} 0 \\ 0 \end{bmatrix}$$

The natural frequencies and mode shapes can then be obtained by solving the eigenvalue problem. Setting up the generalized eigenvalue problem: **these are symmetric M and K matrices so I don't know why  $\tilde{K}$  does not work in a normal eigenvalue problem. Maybe norms of vectors? It is not simple scaling. This example is based off of Rao 6.16 6ed I think it has to do with solving the generalized eigenvalue matrix vs the mass normalized one and when you apply the second  $M^{-1/2}$  as you need it all the time.**

$$K\mathbf{v} = \lambda M\mathbf{v} \quad (165)$$

and solving yields:

$$\lambda_1 = 2.734, \quad \vec{v}_1 = \begin{bmatrix} 0.546819 \\ 0.837251 \end{bmatrix} \quad (166)$$

$$\lambda_2 = 6.765, \vec{v}_2 = \begin{bmatrix} -0.151349 \\ 0.988480 \end{bmatrix}$$

this is than related to the natural frequency and mode shapes as:

$$\omega_1 = \sqrt{\lambda_1} = 1.65 \text{ rad/s}, \vec{v}_1 = \vec{v}_1 \alpha_1 = \begin{bmatrix} 0.546819 \\ 0.837251 \end{bmatrix} \alpha_1 \quad (167)$$

$$\omega_2 = \sqrt{\lambda_2} = 2.60 \text{ rad/s}, \vec{v}_2 = \vec{v}_2 \alpha_2 = \begin{bmatrix} -0.151349 \\ 0.988480 \end{bmatrix} \alpha_2 \quad (168)$$

recall that the eigenvalues only contain information about the direction of the linear transform, and therefore, there magnitudes are arbitrary. Therefore, they must be scaled proportionally to each other. For this reason, the scalars  $\alpha_1$  and  $\alpha_2$  are added. My orthogonalizing the modal vectors with respect to the mass matrix, the values of  $\alpha_1$  and  $\alpha_2$  are found as:

$$1 = \vec{v}_1^T M \vec{v}_1 \quad (169)$$

$$1 = \alpha_1^2 \begin{bmatrix} 0.546819 & 0.837251 \end{bmatrix} \begin{bmatrix} 10 & 0 \\ 0 & 1 \end{bmatrix} \begin{bmatrix} 0.546819 \\ 0.837251 \end{bmatrix} \quad (170)$$

and:

$$1 = \vec{v}_2^T M \vec{v}_2 \quad (171)$$

$$1 = \alpha_2^2 \begin{bmatrix} -0.151349 & 0.98848 \end{bmatrix} \begin{bmatrix} 10 & 0 \\ 0 & 1 \end{bmatrix} \begin{bmatrix} -0.151349 \\ 0.98848 \end{bmatrix} \quad (172)$$

therefore,  $\alpha_1 = 0.520$  and  $\alpha_2 = 0.910$ . Now, applying the proper scaling values to the modal vector, the modal matrix becomes:

$$P = [\vec{v}_1 \ \vec{v}_2] = \begin{bmatrix} 0.284 & -0.137 \\ 0.435 & 0.900 \end{bmatrix} \quad (173)$$

Next, check that the normal modes in the modal matrix ( $P$ ) are normalized, per equation 156. This yields,

$$P^T M P = \begin{bmatrix} 1 & -2.775e-16 \\ -2.775e-16 & 1 \end{bmatrix} \approx I \quad (174)$$

which is close enough to I. Considering that  $\vec{x}(t) = P \vec{q}(t)$ , the EOM for the system can be expressed as:

$$\ddot{\vec{q}}(t) + \begin{bmatrix} \omega_1^2 & 0 \\ 0 & \omega_2^2 \end{bmatrix} \vec{q}(t) = \vec{Q} = \begin{bmatrix} 0 \\ 0 \end{bmatrix} \quad (175)$$

rewriting this in scalar form for each modal coefficient yields:

$$\ddot{q}_i(t) + \omega_i^2 q_i(t) = 0, \ i = 1, 2 \quad (176)$$



where the solution for this ODE is:

$$q_i(t) = q_{i0} \cos(\omega_i t) + \frac{\dot{q}_{i0}}{\omega_i} \sin(\omega_i t) \quad (177)$$

where  $q_{i0}$  and  $\dot{q}_{i0}$  are the initial conditions in modal space. Therefore, the given initial conditions must be transferred into modal space as:

$$\vec{q}(0) = P^T M \vec{x}(0) = \begin{bmatrix} 0.284 & 0.435 \\ -0.137 & 0.900 \end{bmatrix} \begin{bmatrix} 10 & 0 \\ 0 & 1 \end{bmatrix} \begin{bmatrix} 1 \\ 0 \end{bmatrix} = \begin{bmatrix} 2.846 \\ -1.378 \end{bmatrix} \quad (178)$$

$$\dot{\vec{q}}(0) = P^T M \vec{v}(0) = \begin{bmatrix} 0.284 & 0.435 \\ -0.137 & 0.900 \end{bmatrix} \begin{bmatrix} 10 & 0 \\ 0 & 1 \end{bmatrix} \begin{bmatrix} 0 \\ 0 \end{bmatrix} = \begin{bmatrix} 0 \\ 0 \end{bmatrix} \quad (179)$$

therefore,

$$\begin{aligned} q_1(t) &= 2.846 \cdot \cos(1.65t) \\ q_2(t) &= -1.378 \cdot \cos(2.6t) \end{aligned} \quad (180)$$

converting back into the time domain is done knowing  $\vec{x}(t) = P \vec{q}(t)$ , therefore,

$$\vec{x}(t) = P \vec{q}(t) = \begin{bmatrix} 0.284 & -0.137 \\ 0.435 & 0.900 \end{bmatrix} \begin{bmatrix} 2.846 \cdot \cos(1.65t) \\ -1.378 \cdot \cos(2.6t) \end{bmatrix} \quad (181)$$

This further simplified into:

$$\begin{aligned} x_1(t) &= 0.808 \cdot \cos(1.65t) + 0.189 \cdot \cos(2.6t) \\ x_2(t) &= 1.238 \cdot \cos(1.65t) - 1.240 \cdot \cos(2.6t) \end{aligned} \quad (182)$$

These results are plotted in figure 18.

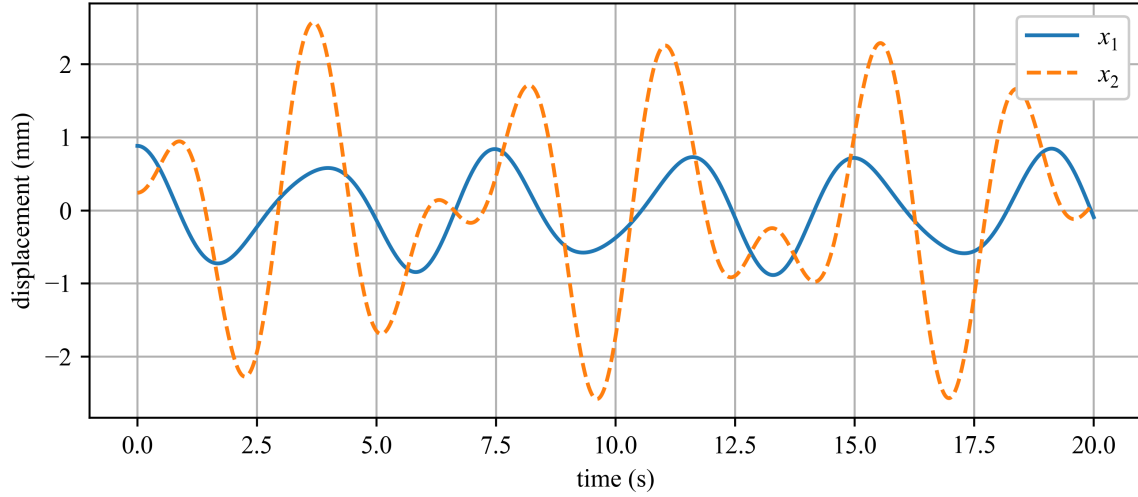


Figure 18: Temporal response for the 2-DOF reconstructed using just all the modal coordinates.

Next, the truncated response can be computed by only considering the first mode response for the system (i.e.  $\vec{x}(t) = q_1(t)\vec{v}_1$ ). This is obtain as:

$$\vec{x}(t) = P\vec{q}_{\text{truncated}}(t) = \begin{bmatrix} 0.284 & -0.137 \\ 0.435 & 0.900 \end{bmatrix} [2.846 \cdot \cos(1.65t)] \quad (183)$$

This further simplified into:

$$\begin{aligned} x_1(t) &= 0.808 \cdot \cos(1.65t) \\ x_2(t) &= 1.238 \cdot \cos(1.65t) \end{aligned} \quad (184)$$

These results are plotted in figure 19. Note that this only considers the response of the system that is a function of the first mode. Note that this captures some of the “general” idea of the system while missing out on the finer points that the 2<sup>nd</sup> mode contributes.

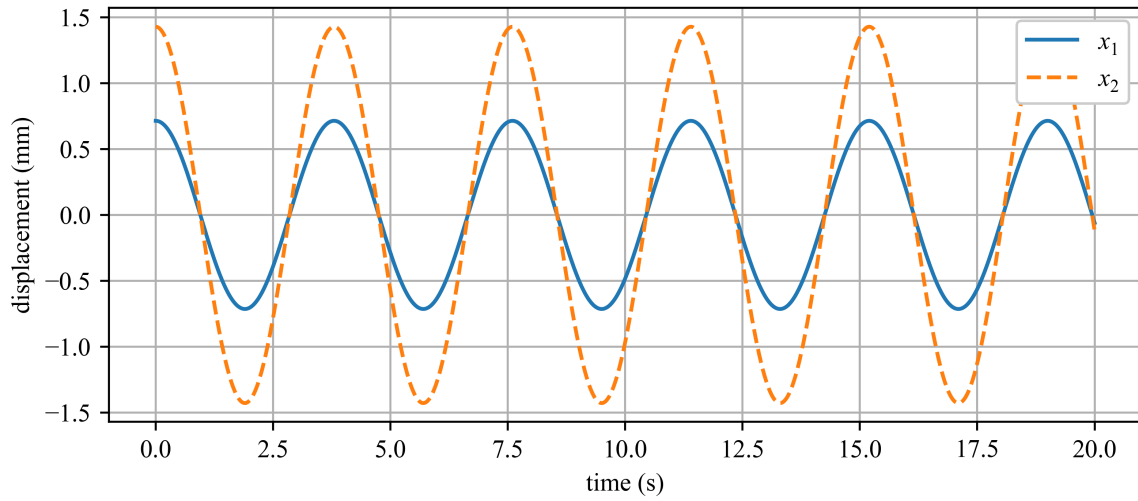


Figure 19: Truncated temporal response for the 2-DOF reconstructed using just the first modal coordinates.

Lastly, the participation of the two modes can be plotted from the time series response of equation 181.

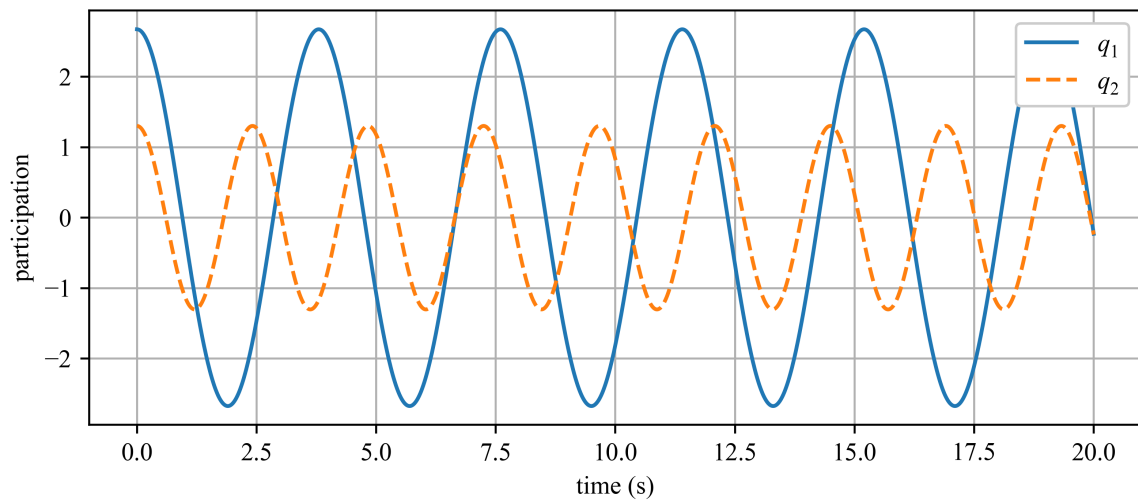


Figure 20: Modal participation coefficients.