

AeroCats_2014

$\alpha = 9.0000$ $pb/2V = 0.0000$ $CL = 1.4903$ $Cl = -0.0000$
 $\beta = 0.0000$ $qc/2V = 0.0000$ $CY = 0.0000$ $Cm = -0.0451$
 $M = 0.000$ $rb/2V = 0.0000$ $CD = 0.16515$ $Cn = 0.0000$
Elevator = 0.0000 $CD_i = 0.13792$ $e = 1.0830$
Aileron = 1 $CD_p = 0.00000$

C_{ℓ_I}
 C_{ℓ}
 $C_{\ell} C / C_{ref}$
 α_i

