

AeroCats_2014

$\alpha = 9.0000$ $pb/2V = 0.0000$ $CL = 1.5526$ $Cl = -0.0000$
 $\beta = 0.0000$ $qc/2V = 0.0000$ $CY = -0.0000$ $Cm = 0.0104$
 $M = 0.000$ $rb/2V = 0.0000$ $CD = 0.16606$ $Cn = 0.0000$
Elevator = 0.0000 $CD_i = 0.16450$ $e = 0.9778$
Ailero $CD_p = 0.00000$

C_{l_I}
 C_l
 $C_l C / C_{ref}$
 α_i

