

AeroCats_2014

α = 9.0000 pb/2V = 0.0000 CL = 1.5641 Cl = -0.0000
 β = 0.0000 qc/2V = 0.0000 CY = 0.0000 Cm = 0.0109
M = 0.000 rb/2V = 0.0000 CD = 0.16588 Cn = 0.0000
Elevator = 0.0000 CD_i = 0.16032 e = 1.0148
Aileron CD_p = 0.00000

$C_{\ell I}$
 C_{ℓ}
 $C_{\ell} C / C_{ref}$
 α_i

