

Xoptfoil-JX is a modified version of Xoptfoil version 1.11.1 – the airfoil optimizer by Daniel Prosser

Xoptfoil-JX Reference

Jochen Guenzel 2019

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- see 'Xoptfoil-JX Description' for more detailed information.

Aerodynamic target values

With the new optimization types 'target-drag' and 'target-moment' one can define a certain 'target_value' for an operating point. With this the Xoptfoil optimization tries to get to the value of the design variable as close as possible to the defined target value. As with the normal optimization types the gravitational force towards the target value is controlled by the weighting of the operating point.

<code>optimization_type(n)</code>	Additional types 'target-drag' and 'target-moment'
<code>target_value(n)</code>	Numerical value which should be achieved. A negative value will be taken to multiply the value of the seed airfoil to get the target value – so "-1.0" would mean: Achieve the value of the seed airfoil.

Example:

```
op_mode(7) = 'spec-cl'  
op_point(7) = 0.8  
optimization_type(7) = 'target-drag'  
target_value(7) = 0.0142  
reynolds(7) = 146E+03  
weighting(7) = 0.7
```

Geometric target values

If there is the need that the final airfoil should have a certain thickness it is more natural for the optimization process to define the thickness as an optimization target instead being a constraint.

Another use case is to have control over a sometimes bad deformation of the surface on the upper or lower side of the airfoil especially towards the trailing edge.

<code>&geometric_targets</code>	New namelist
<code>ngeotargets</code>	Number of geometric targets that shall be applied
<code>target_type(n)</code>	Type of geometric target – either 'zBot' z-value at the bottom side 'zTop' z-value on the top side 'Thickness' final thickness of the airfoil
<code>target_geo(n)</code>	Numerical value which should be achieved. A negative value will be taken to multiply the value of the seed airfoil to get the target value – so "-1.0" would mean: Achieve the value of the seed airfoil.
<code>x_Pos(n)</code>	X-Position along the chord for this geometric target. Only needed for 'zBot' and 'zTop'
<code>weighting_geo(n)</code>	Weighting of this target within the objective function. Same meaning as 'weighting(N)' for an operating point

Example

```
&geometric_targets  
ngeotargets = 1
```

```

x_Pos(1) = 0.7
target_type(1) = 'zBot'
target_geo(1) = 0.012
weighting_geo(1) = 0.5

```

New shape_functions 'camb-thick'

Beside the shape_functions type 'naca' and 'hicks-henne' there is a new type implemented called 'camb-thick'. In this case the seed airfoil will be modified during optimization by the 4 airfoil parameters

- thickness
- x-location of max. thickness
- camber
- x-location of max. camber

The airfoil modification is done by the two xfoil routines THKCAM and HIPNT

&optimization options	high-level optimization and parameterization parameters are
shape_functions	= 'naca'
	= 'hicks-henne'
	= 'camb-thick' new

Additional command line options options

-r xxxxxxxx	Allows the definition of a default reynolds number xxxxxxxx for the operating points. This value will be taken if reynolds() is not defined for an operating point. With this option a single inputs.txt can be used for different polar based optimizations.
-a airfoil_file	Define the filename of the seed airfoil. A 'airfoil_file' within inputs.txt will be overwritten.

Additional display options

&optimization options	high-level optimization and parameterization parameters are
show_details	= .true. show more detailed information during optimization (default)
	= .false do not show
echo_input_parms	= .true. echo all input parameter (default)
	= .false suppress echoing

Additional operating options

<code>&operating_conditions</code>	common parameters for all operating points
<code>re_default</code>	<code>= xxxxxxxx</code> Default value for reynolds number which is taken if a reynolds(i) for an operating point is not defined. This is useful if all operating points do have the same reynolds number – see also command line argument ‘-r’ (equals to <code>re_default</code>)
<code>re_default_as_resqrtcl</code>	<code>= .true.</code> take <code>re_default</code> as the value of <code>re*sqrt(cl)</code> which equals to the Type 2 polar. This is convenient if you have Type 2 based operating points <code>= .false</code> <code>re_default</code> is taken as it is (default)
<code>optimization_type()</code>	<code>= 'min-lift-slope'</code> Minimize slope of <code>cl(alpha)</code> curve at operating point. This type can be used to get <code>cl-max</code> at a certain angle of attack. Take a little care when using this option: There should be a helper (dummy) operating point to the left and to the right with a distance of e.g. 0.5 degrees. These helper points are needed to calculate the slope at the current operating point. <code>= 'min-glide-slope'</code> Minimize slope of glide ration (<code>cl/cd</code>)/ <code>cl</code> curve at operating point. This type can be used to get the best glide ration at a certain <code>cl</code> value.. Take a little care when using this option: There should be a helper (dummy) operating point to the left and to the right with a delta <code>cl</code> of 0.1. These helper points are needed to calculate the slope at the current operating point.

Generation of polars of the optimized airfoil

This option allows to generate a polar set in xfoil-format at the end of the optimization. This is especially useful, if further analysis of the airfoil will be done in xflr5 or flow5.

The polars will be stored in the subdirectory `<airfoil-name>_polars.`

<code>&polar_generation</code>	Namelist to define the polar set
<code>generate_polars</code>	<code>= .true.</code> Polars as defined in this namelist will be generated at the end of the optimization <code>= .false</code> (default) No polars will be generated
<code>type_of_polar</code>	<code>= 1</code> Type 1 polars (fixed speed) will be generated <code>= 2</code> Type 2 polars (fixed lift) will be generated. In this case the specified reynolds number will be taken as <code>RE*sqrt(cl)</code>
<code>polar_reynolds</code>	<code>= xxxxxx, yyyyyy, zzzzzz, ...</code> List of reynolds numbers for the polars to be generated
<code>op_mode</code>	<code>= 'spec-al'</code> The operating points of the polars specified with <code>op_point_range</code> are based on alpha (aoa) <code>= 'spec-cl'</code> The operating points of the polars specified with <code>op_point_range</code> are based on cl
<code>op_point_range</code>	<code>= <start>, <end>, <increment></code> Defines the series of operating points of a polar. For <code>op_mode = 'spec-al'</code> this could be for example: <code>-3.0, 10.0, 0.5</code>

Airfoils surface quality and smoothing

Activate a simple smoothing algorithm to get rid off micro waves (spikes) in the surface of the (seed) airfoil.

<code>&smoothing_options</code>	New namelist
<code>do_smoothing</code>	= <code>.true.</code> - activate smoothing of the surface
<code>show_smoothing</code>	= <code>.true.</code> - show additional informations during the optimization process
<code>highlow_treshold</code>	Numerical value to detect highs and lows of the 2 nd derivative of the surface curve. The number of high and lows is taken into account to assess the quality of the surface. A good value to start is 0.05
<code>spike_threshold</code>	Numerical value to detect highs and lows of the 3 rd derivative called spikes. The number of spikes is taken into account to assess the quality of the surface. A good value to start is 2
<code>weighting_smoothing</code>	Out of the number of highs, lows and spikes a "perturbation value" is calculated which can be an indication for the surface quality: '0' is excellent, a value greater '1' indicate some problems... If this value should be part of the objective function 'weighting_smoothing' should have a value > 0.0.

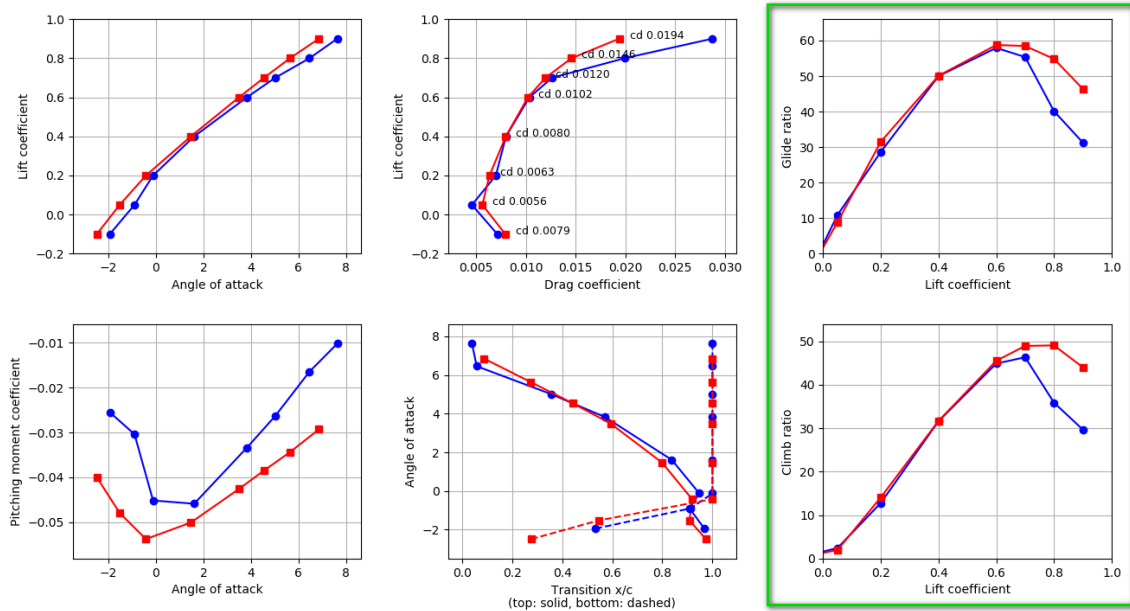
Example

```
&smoothing_options
do_smoothing      = .true.
show_smoothing    = .true.
highlow_treshold  = 0.04
spike_threshold   = 1.5
weighting_smoothing = 0.1
```

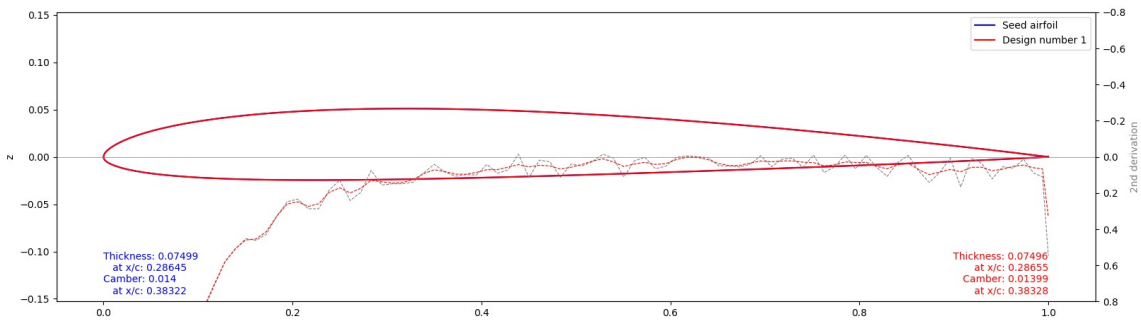
xoptfoil_visualizer-jx.py - Modifications of the Visualizer -

To get an earlier indication whether the optimized airfoil will have the desired aerodynamic properties the polar window of the visualizer is complemented by two sub views for glide ratio and climb ratio.

The layout is optimized for a monitor resolution of 1920 x 1080.



To visualize the smoothing process and to get information about the surface quality additional information can be displayed in the airfoil window:



- 2nd derivation of the airfoil surface
- 3rd derivation of the airfoil surface
- z-values applied by xoptfoil (sum of all shape functions)

This is done by setting these variables in the Python script

```
490 plot_2nd_deriv = True
491 plot_3rd_deriv = False
492 plot_delta_y = True
```

Have fun!