



A318/A319/A320/A321

SERVICE BULLETIN SUMMARY

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This summary is for information only
and is not approved for modification of the aircraft

ATA SYSTEM : 21

TITLE : AIR CONDITIONING - AVIONICS EQUIPMENT GROUND COOLING - DEACTIVATE THE
GROUND REFRIGERATION UNIT (GRU).

MODIFICATION No. : 33129P8139

REASON/DESCRIPTION/OPERATIONAL CONSEQUENCES

Due to restrictions imposed by the scope of Montreal protocol, the refrigerant used in the Ground Refrigeration Unit (GRU) PN 786B010000 is no longer allowed to be used for the production or maintenance of refrigeration units.

Operators are therefore required to either upgrade their GRU to the standard PN 799A010000 (RFC/RMO applies) or de-activate it.

For the operators who do not wish to upgrade the GRU to the standard PN 799A010000, this Service Bulletin consists in de-activating the GRU while leaving it in-situ.

Accomplishment of this Service Bulletin will allow the operators to comply with the content of the Montreal protocol.

Accomplishment of this Service bulletin will reduce the maintenance tasks.

Maintenance Planning Document (MPD) No. 212700-01-1, 212700-02-1, 212700-03-1, 212700-04-1, 212700-05-1 and 212742-01-1 must be deleted after accomplishment of this Service Bulletin.

EVALUATION TABLE			
COMPLIANCE	Desirable	CANCELS INSPECTION SB	No
POTENTIAL AD	No	A/C OPERATION AFFECTED	Yes
RELIABILITY AFFECTED	No	PAX COMFORT AFFECTED	No
COST SAVING	No	ETOPS AFFECTED	No
STRUCTURAL LIFE EXTN	No	VENDOR SB INVOLVED	No
KIT PRICE (USD)	See SB		

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EFFECTIVITY

This Service Bulletin is applicable to these operators :

A1L ALK CTN CYP D2F IAC KAC MSR MXA RJA TAR TNA

CONCURRENT REQUIREMENTS

None

REFERENCES/REPERCUSSIONS

TFU	: None
OEB	: None
AOT	: None
SIL	: None
LIFE LIMIT	: None
LINE MAINTENANCE AFFECTED	: No
OTHERS	: None

NATURE OF THE WORK

AIRCRAFT	: YES
EQUIPMENT	: NO
HARD	: NO
SOFT	: NO
OBRM	: NO

MANPOWER

TOTAL MANHOURS	22.5
ELAPSED TIME (HOURS)	15.0

MATERIAL INFORMATION

AIRCRAFT DATA

Kit 211148A01R01

Plug assy

Kit 211148A02R00

thru

Kit 211148A11R00

Integrally lighted plate

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APPENDICES

None

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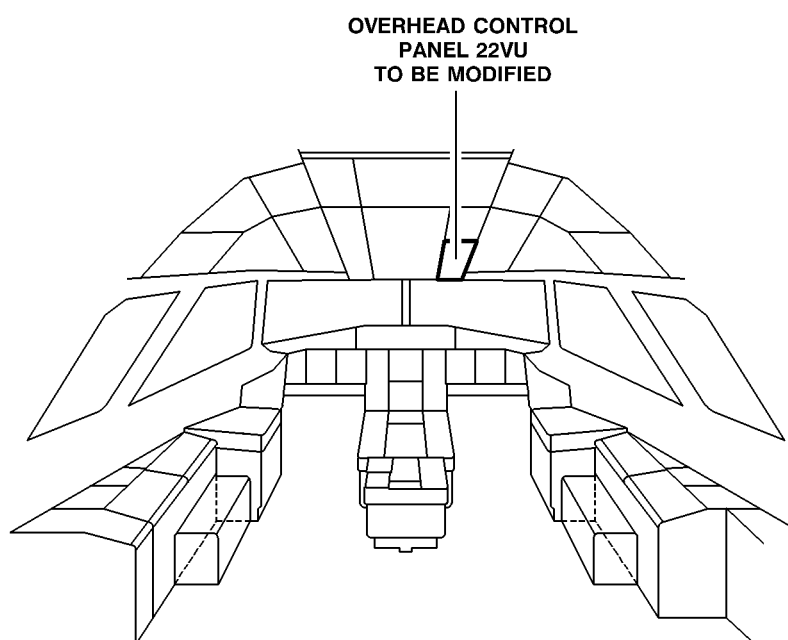
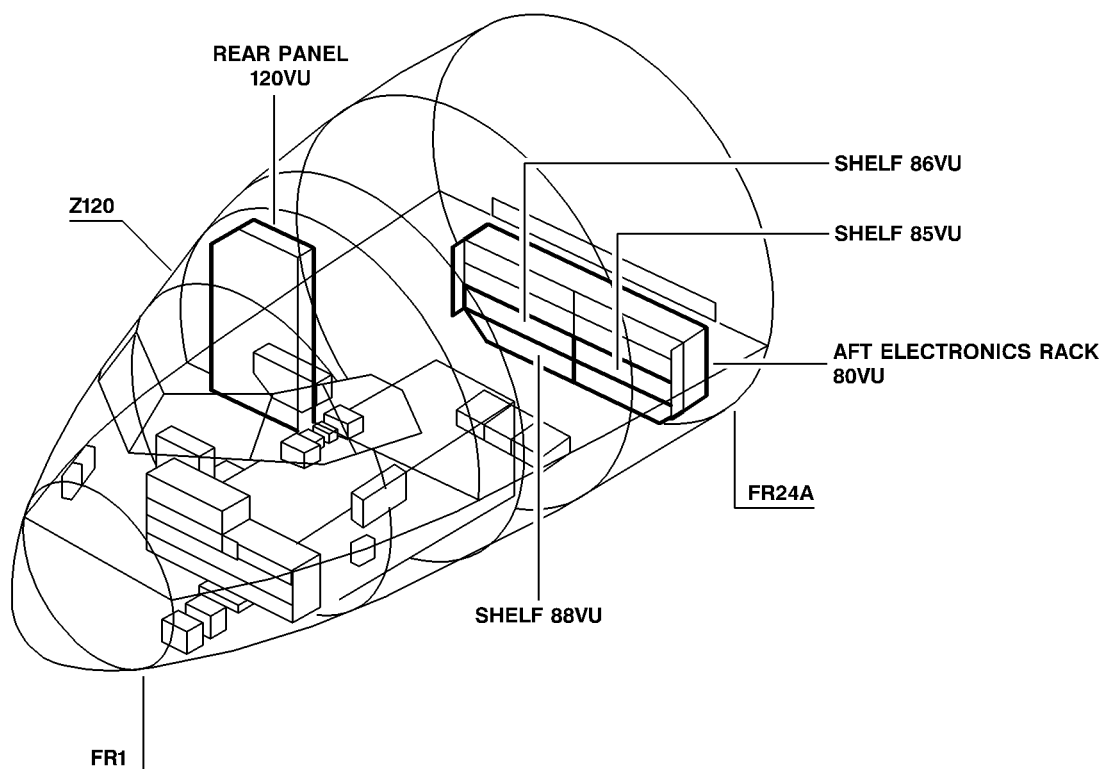
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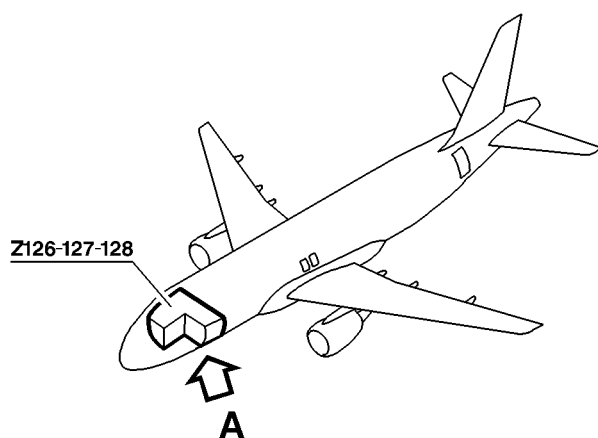
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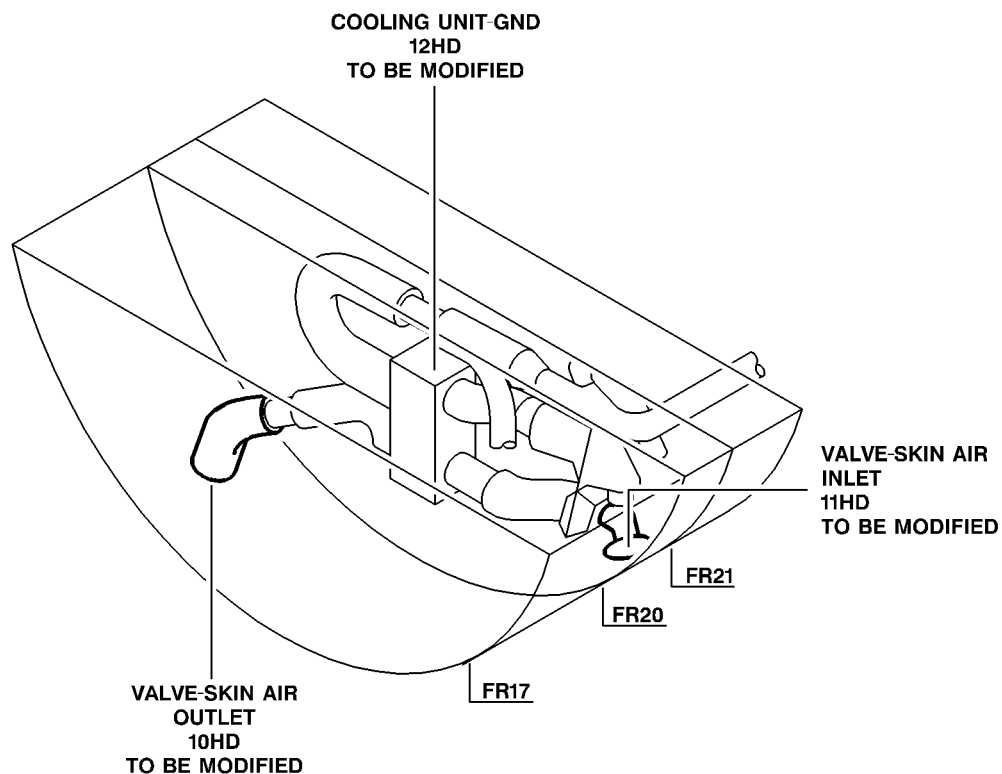


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ATA SYSTEM : 21

TITLE : AIR CONDITIONING - AVIONICS EQUIPMENT GROUND COOLING - DEACTIVATE THE
GROUND REFRIGERATION UNIT (GRU).

MODIFICATION No. : 33129P8139

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1. PLANNING INFORMATION

A. EFFECTIVITY

(1) Models

320-211 320-212 320-231 320-232 321-131 321-231

(2) Aircraft

(a) Effectivity by MSN

This Service Bulletin is applicable to aircraft MSN :

0028 0035 0037 0038 0045-0051 0056-0058 0074 0075 0080
0087-0090 0095-0097 0123 0165 0166 0178 0180-0182 0194 0195
0198 0256 0258 0295 0316 0322 0326 0332 0344 0351 0366 0369
0396 0398 0406 0416 0423 0431 0432 0451 0469 0486 0490 0492
0499 0538 0555 0569 0606 0680 0687 0715 0725 0731 0746 0791
0822

NOTE (01) This Service Bulletin is only applicable to aircraft on which Mod. No. 20056P0073 (AIR CONDITIONING - AVIONICS EQUIPMENT - GROUND COOLING - INSTALL COOLING SYSTEM) is embodied or Service Bulletin No. A320-21-1067) is accomplished.

NOTE (02) This modification is applicable by Service Bulletin only.

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(b) Effectivity by Operator

The Operator/MSN relationship is provided for information only and is correct at the time of issue in accordance with the information available to AIRBUS. Any future changes resulting from transfer of an aircraft from one operator to another will not be reflected in this list unless the Service Bulletin is revised for another reason.

OPERATOR	MSN
A1L	0322
ALK	0406
CTN	0258
CYP	0028 0035 0037 0038 0180 0256 0295 0316
D2F	0326 0344
IAC	0045 0046 0047 0048 0049 0050 0051 0056 0057 0058 0074 0075 0080 0089 0090 0095 0096 0097 0396 0398 0416 0423 0431 0432 0451 0469 0486 0490 0492 0499
KAC	0181 0182 0195
MSR	0165 0166 0178 0194 0198 0351 0366 0680 0687 0715 0725
MXA	0332 0369
RJA	0087 0088 0569
TAR	0123
TNA	0538 0555 0606 0731 0746 0791 0822

(c) Effectivity by MSN and Kit/Configuration

The kits and configurations applicable to these MSNs are given at the end of this list:
0028 0035 0037-0038

KIT No.	QTY	PER A/C	CONFIGURATION
211148A01R01	1	01	(01)
211148A05R00	1	None	(06)

NOTE (01) Config. 01 concerns the aircraft on which modification No. 21420P1641 (LIGHTS - WIRING PROVISIONS FOR OPTIONAL LIGHT ON TRANSFORMER 37LP-FU SE 6 IN 22VU) is not embodied.

NOTE (06) 211148A05R00 concerns the modification of the integrally lighted plate (9LF) for aircraft on which old PN D33110086A00 is installed.

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The kits and configurations applicable to these MSNs are given at the end of this list:

0045-0051 0056-0058 0074-0075 0080

KIT No.	QTY PER A/C	CONFIGURATION
211148A01R01	1	01 (01)
211148A06R00	1	None (07)
211148A08R00	1	None (08)

NOTE (01) Config. 01 concerns the aircraft on which modification No. 21420P1641 (LIGHTS - WIRING PROVISIONS FOR OPTIONAL LIGHT ON TRANSFORMER 37LP-FU SE 6 IN 22VU) is not embodied.

NOTE (07) 211148A06R00 concerns the modification of the integrally lighted plate (9LF) for aircraft on which old PN D33110137A00 is installed.

NOTE (08) 211148A08R00 concerns the modification of the integrally lighted plate (9LF) for aircraft on which old PN D33110087A00 is installed.

NOTE : Selection of Kit 211148A06R00 or Kit 211148A08R00 depends on the existing integrally lighted plate (9LF) on the overhead control panel (22VU).

The kits and configurations applicable to these MSNs are given at the end of this list:

0332 0369 0538 0555 0606 0731 0746 0791 0822

KIT No.	QTY PER A/C	CONFIGURATION
211148A01R01	1	02 (02)
211148A02R00	1	None (03)

NOTE (02) Config. 02 concerns the aircraft on which modification No. 21420P1641 (LIGHTS - WIRING PROVISIONS FOR OPTIONAL LIGHT ON TRANSFORMER 37LP-FU SE 6 IN 22VU) is embodied.

NOTE (03) 211148A02R00 concerns the modification of the integrally lighted plate (9LF) for aircraft on which old PN D33110123B00 is installed.

The kits and configurations applicable to these MSNs are given at the end of this list:

0087-0088 0123 0181-0182 0195 0258 0569

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KIT No.	QTY PER A/C CONFIGURATION		
211148A01R01	1	02	(02)
211148A03R00	1	None	(04)

NOTE (02) Config. 02 concerns the aircraft on which modification No. 21420P1641 (LIGHTS - WIRING PROVISIONS FOR OPTIONAL LIGHT ON TRANSFORMER 37LP-FU SE 6 IN 22VU) is embodied.

NOTE (04) 211148A03R00 concerns the modification of the integrally lighted plate (9LF) for aircraft on which old PN D33110104A00 is installed.

The kits and configurations applicable to these MSNs are given at the end of this list:

0165-0166 0178 0194 0198 0351 0366

KIT No.	QTY PER A/C CONFIGURATION		
211148A01R01	1	02	(02)
211148A04R00	1	None	(05)

NOTE (02) Config. 02 concerns the aircraft on which modification No. 21420P1641 (LIGHTS - WIRING PROVISIONS FOR OPTIONAL LIGHT ON TRANSFORMER 37LP-FU SE 6 IN 22VU) is embodied.

NOTE (05) 211148A04R00 concerns the modification of the integrally lighted plate (9LF) for aircraft on which old PN D33110110A00 is installed.

The kits and configurations applicable to these MSNs are given at the end of this list:

0180 0256 0295 0316 0406

KIT No.	QTY PER A/C CONFIGURATION		
211148A01R01	1	02	(02)
211148A05R00	1	None	(06)

NOTE (02) Config. 02 concerns the aircraft on which modification No. 21420P1641 (LIGHTS - WIRING PROVISIONS FOR OPTIONAL LIGHT ON TRANSFORMER 37LP-FU SE 6 IN 22VU) is embodied.

NOTE (06) 211148A05R00 concerns the modification of the integrally lighted plate (9LF) for aircraft on which old PN D33110086A00 is installed.

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The kits and configurations applicable to these MSNs are given at the end of this list:
0486 0490 0492 0499

KIT No.	QTY PER A/C CONFIGURATION		
211148A01R01	1	02	(02)
211148A06R00	1	None	(07)

NOTE (02) Config. 02 concerns the aircraft on which modification No. 21420P1641 (LIGHTS - WIRING PROVISIONS FOR OPTIONAL LIGHT ON TRANSFORMER 37LP-FU SE 6 IN 22VU) is embodied.

NOTE (07) 211148A06R00 concerns the modification of the integrally lighted plate (9LF) for aircraft on which old PN D33110137A00 is installed.

The kits and configurations applicable to these MSNs are given at the end of this list:
0089-0090 0095-0097 0396 0398 0416 0423 0431-0432 0451 0469

KIT No.	QTY PER A/C CONFIGURATION		
211148A01R01	1	02	(02)
211148A06R00	1	None	(07)
211148A08R00	1	None	(08)

NOTE (02) Config. 02 concerns the aircraft on which modification No. 21420P1641 (LIGHTS - WIRING PROVISIONS FOR OPTIONAL LIGHT ON TRANSFORMER 37LP-FU SE 6 IN 22VU) is embodied.

NOTE (07) 211148A06R00 concerns the modification of the integrally lighted plate (9LF) for aircraft on which old PN D33110137A00 is installed.

NOTE (08) 211148A08R00 concerns the modification of the integrally lighted plate (9LF) for aircraft on which old PN D33110087A00 is installed.

NOTE : Selection of Kit 211148A06R00 or Kit 211148A08R00 or Kit 211148A11R00 depends on the existing integrally lighted plate (9LF) on the overhead control panel (22VU).

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The kits and configurations applicable to these MSNs are given at the end of this list:
0680 0687 0715 0725

KIT No.	QTY	PER A/C	CONFIGURATION
211148A01R01	1	02	(02)
211148A07R00	1	None	(09)

NOTE (02) Config. 02 concerns the aircraft on which modification No. 21420P1641 (LIGHTS - WIRING PROVISIONS FOR OPTIONAL LIGHT ON TRANSFORMER 37LP-FU SE 6 IN 22VU) is embodied.

NOTE (09) 211148A07R00 concerns the modification of the integrally lighted plate (9LF) for aircraft on which old PN D33110197A00 is installed.

The kits and configurations applicable to these MSNs are given at the end of this list:
0322 0326 0344

KIT No.	QTY	PER A/C	CONFIGURATION
211148A01R01	1	02	(02)
211148A09R00	1	None	(10)
211148A10R00	1	None	(11)
211148A11R00	1	None	(12)

NOTE (02) Config. 02 concerns the aircraft on which modification No. 21420P1641 (LIGHTS - WIRING PROVISIONS FOR OPTIONAL LIGHT ON TRANSFORMER 37LP-FU SE 6 IN 22VU) is embodied.

NOTE (10) 211148A09R00 concerns the modification of the integrally lighted plate (9LF) for aircraft on which old PN D33110141B00 is installed.

NOTE (11) 211148A10R00 concerns the modification of the integrally lighted plate (9LF) for aircraft on which old PN D33110271A00 is installed.

NOTE (12) 211148A11R00 concerns the modification of the integrally lighted plate (9LF) for aircraft on which old PN D00410458A00 is installed.

NOTE : Selection of Kit 211148A09R00 or Kit 211148A10R00 or Kit 211148A11R00 depends on the existing integrally lighted plate (9LF) on the overhead control panel (22VU).

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(3) Spares

Kit 211148A01R01

Skin Air Valve VFT210

or

Skin Air Valve VFT210A

or

Skin Air Valve VFT210-1

Kit 211148A02R00

Integrally Lighted Plate D33110123B00

Kit 211148A03R00

Integrally Lighted Plate D33110104A00

Kit 211148A04R00

Integrally Lighted Plate D33110110A00

Kit 211148A05R00

Integrally Lighted Plate D33110086B00

Kit 211148A06R00

Integrally Lighted Plate D33110137A00

Kit 211148A07R00

Integrally Lighted Plate D33110197A00

Kit 211148A08R00

Integrally Lighted Plate D33110087A00

Kit 211148A09R00

Integrally Lighted Plate D33110141B00

Kit 211148A10R00

Integrally Lighted Plate D33110271A00

Kit 211148A11R00

Integrally Lighted Plate D00410458A00

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B. CONCURRENT REQUIREMENTS

None

C. REASON

(1) History

Due to restrictions imposed by the scope of Montreal protocol, the refrigerant used in the Ground Refrigeration Unit (GRU) PN 786B010000 is no longer allowed to be used for the production or maintenance of refrigeration units.

Operators are therefore required to either upgrade their GRU to the standard PN 799A010000 (RFC/RMO applies) or de-activate it.

(2) Objective/Action

For the operators who do not wish to upgrade the GRU to the standard PN 799A010000, this Service Bulletin consists in de-activating the GRU while leaving it in-situ.

(3) Advantages

Accomplishment of this Service Bulletin will allow the operators to comply with the content of the Montreal protocol.

(4) Operational/Maintenance Consequences

Accomplishment of this Service bulletin will reduce the maintenance tasks.

Maintenance Planning Document (MPD) No. 212700-01-1, 212700-02-1, 212700-03-1, 212700-04-1, 212700-05-1 and 212742-01-1 must be deleted after accomplishment of this Service Bulletin.

D. DESCRIPTION

To accomplish this Service Bulletin it is necessary to :

(1) For Kit 211148A01R01

- (a) In the avionics compartment, LH side, modify the inlet skin air valve (11HD).
- (b) In the avionics compartment, RH side, Modify the outlet skin air valve (10HD).
- (c) Modify the connectors of the fan-GND cool (8HD), the cooling unit-GND (12HD) and press SW-GND cool fan (13HD), in the avionics compartment.
- (d) Modify the wiring on shelf 88VU, in the avionics compartment.

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- (e) Modify the pin programming and the parity of the SDAC on the shelf 85VU and the shelf 86VU, in the avionics compartment.
- (f) Modify the equipment in the rear circuit breaker panel 122VU, in the cockpit.
- (g) Modify the equipment in the power center-AC 123VU, in the cockpit.
- (2) Kit 211148A02R00 , modify the equipment and the wiring in the overhead control panel 22VU, in the cockpit.
- (3) Kit 211148A03R00 , modify the equipment and the wiring in the overhead control panel 22VU, in the cockpit.
- (4) Kit 211148A04R00 , modify the equipment and the wiring in the overhead control panel 22VU, in the cockpit.
- (5) Kit 211148A05R00 , modify the equipment and the wiring in the overhead control panel 22VU, in the cockpit.
- (6) Kit 211148A06R00 , modify the equipment and the wiring in the overhead control panel 22VU, in the cockpit.
- (7) Kit 211148A07R00 , modify the equipment and the wiring in the overhead control panel 22VU, in the cockpit.
- (8) Kit 211148A08R00 , modify the equipment and the wiring in the overhead control panel 22VU, in the cockpit.
- (9) Kit 211148A09R00 , modify the equipment and the wiring in the overhead control panel 22VU, in the cockpit.
- (10) Kit 211148A10R00 , modify the equipment and the wiring in the overhead control panel 22VU, in the cockpit.
- (11) Kit 211148A11R00 , modify the equipment and the wiring in the overhead control panel 22VU, in the cockpit.

E. COMPLIANCE

(1) Classification

Desirable

(2) Accomplishment Timescale

In accordance with operators' maintenance schedule.

F. APPROVAL

The technical content of this Service Bulletin has been approved under the authority of the DGAC Design Organisation Approval No. F.JA.02.

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If an aircraft listed in the effectivity has a modification or repair embodied that is not of AIRBUS origin, and which affects the content of this Service Bulletin, the operator is responsible for obtaining approval by its airworthiness authority for any adaptation necessary before incorporation of the Service Bulletin.

G. MANPOWER

The manpower estimates given in this Service Bulletin are based on the direct labor cost to do the work. These estimates assume that the work will be done by experienced personnel, and may need to be revised upwards to suit operator's circumstances. The estimates do not include the time to prepare, plan or inspect the work. Manufacture and procurement of parts and tools, drying times for paints, sealants, etc, and general administration work are also not included.

Kit 211148A01R01 thru Kit 211148A11R00

Get access	2.0
Modif. of the skin air valve (11HD)	1.5
Modif. of the skin air valve (10HD)	1.5
Modif. of the connectors in the AV/COMP	2.0
Modif. of the pin programming	1.0
Modification of the parity	0.5
Modif. of the C/B panel 122VU	1.0
Modif. of the center-AC 123VU	1.0
Modif. of the wiring in the panel 22VU	2.0
Modif. of the equipment on the 22VU	2.0
Modification of the integrally plate 9LF	1.0
Test	5.0
Close-up	2.0
TOTAL MANHOURS	22.5
ELAPSED TIME (HOURS)	15.0

H. WEIGHT AND BALANCE

Kit 211148A01R01 thru Kit 211148A11R00

Manufacturers Empty Weight : -0.120 kg (-0.2645 lb)

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Effect on Balance : -0.570 kgm (-4.122 lb.ft)

I. ELECTRICAL LOAD DATA

(1) Direct Current (DC) Load Changes

Kit 211148A01R01 thru Kit 211148A11R00

Designation : AIR COND/VENT/GND/COOL/CTL

Circuit breaker : 3HD (Existing)

Busbar : 103PP

Electrical Load Data : -90 W

(2) Alternating Current (AC) Load Changes

Kit 211148A01R01 thru Kit 211148A11R00

Designation : VENT/GND COOL/UNIT

Circuit breaker : 1HD (Existing)

Bus-Bar :1XP/A/B/C

Electrical Load Data : -1827 VA

Designation : VENT/GND COOL/UNIT

Circuit breaker : 2HD (Existing)

Electrical Load Data : -1827 VA

Bus-Bar :2XP/A/B/C

J. REFERENCES

Aircraft Maintenance Manual (AMM) : 12-34-24 20-21-11 21-21-00
21-26-00 21-27-00 21-27-52
21-27-53 21-43-00 22-96-00
22-97-00 23-26-00 24-38-51
26-16-00 26-23-00 30-45-00
31-50-00 31-55-34 31-60-00
33-13-00 33-14-00 52-41-00
73-22-34

Consumable Material List (CML)

Elec. Std. Practices Manual (ESPM) : 20-33-00 20-51-10 20-55-00

Maintenance Planning Document (MPD) : 212700-01-1 212700-02-1
212700-03-1 212700-04-1
212700-05-1 212742-01-1

Standards Manual (SM)

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Structural Repair Manual (SRM) : 51-40-00 51-43-12 51-44-11
51-46-11

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K. PUBLICATIONS AFFECTED

Aircraft Maintenance Manual (AMM) : 05-53-00 21-27-00 21-27-34
21-27-42 21-27-51 21-27-52
21-27-53

Aircraft Schematic Manual (ASM) : 21-27-01 31-54-02

Aircraft Wiring List (AWL)

Aircraft Wiring Manual (AWM) : 21-27-01 21-27-02 21-27-03
21-27-04 21-27-05 31-54-05
31-54-06 31-54-29 33-14-02
93-00-22 93-00-88 93-01-22
93-01-23 94-20-20 94-23-05
94-23-08 94-30-00

Illustrated Parts Catalog (IPC) : 21-27-01 21-27-08 33-13-08

Trouble Shooting Manual (TSM) : 21-27-00

Master Minimum Equipment List (MMEL)

Flight Crew Operating Manual (FCOM) : VOL 1 VOL 3

This Service Bulletin has an operational impact. You are therefore requested to inform AIRBUS, at least four weeks in advance, of your intention to accomplish this Service Bulletin, in order to receive required amendments to your operational documentation. For this purpose, please send a fax, telex or mail to :

AI/SE-D, AIRBUS - Customer Services Directorate
Fax No. (33) 5 61 93 28 06
Sita No. TLSBP7X
E-mail sb.reporting@airbus.fr

These amendments will be confirmed by the next normal revision of operational documentation, once your Service Bulletin accomplishment report has been sent, as usual, to our Customer Services Directorate (SE-D).

L. INTERCHANGEABILITY/MIXABILITY

DESCRIPTION	OLD PART No.	NEW PART No.	INT	MIXABILITY
Integrally Lighted Plate	D33110104A00	D33110046A00	03	Not applicable
Integrally Lighted Plate	D33110141B00	D33110084A00	03	Not applicable
Integrally Lighted Plate	D33110271A00	D33110120A00	03	Not applicable

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DESCRIPTION	OLD PART No.	NEW PART No.	INT	MIXABILITY
Integrally Lighted Plate	D00410458A00	D33110121A00	03	Not applicable
Integrally Lighted Plate	D33110123B00	D33110183A00	03	Not applicable
Integrally Lighted Plate	D33110110A00	D33110229A00	03	Not applicable
Integrally Lighted Plate	D33110087A00	D33110255A00	03	Not applicable
Integrally Lighted Plate	D33110086B00	D33110284A00	03	Not applicable
Integrally Lighted Plate	D33110137A00	D33110285A00	03	Not applicable
Integrally Lighted Plate	D33110197A00	D33110286A00	03	Not applicable

NOTE : For definitions of interchangeability codes shown in column INT.
refer to ATA. Common Support Data Dictionary (CSDD), Chapter 2.

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2. MATERIAL INFORMATION

A. MATERIAL – PRICE AND AVAILABILITY

(1) Material

Customers with aircraft shown in the effectivity of this Service Bulletin should send a purchase order to AIRBUS. Quote the number of this Service Bulletin. The address is :

AIRBUS
SPARES SUPPORT AND SERVICES
P.O. Box 630262
22312 HAMBURG
GERMANY

(2) Price and Availability

Kit 211148A01R01

Cost : 1260 US Dollars

Availability : 99 calendar days from receipt of order

The Kit availability given above is the standard lead time from the date of your purchase order. If you require the Kit(s) before this time, please include a retrofit planning schedule with your order so that we can try to comply with your requirements.

The sales terms (costs and availability) are estimated in relation to economic conditions at the issue date of the Service Bulletin.

Kit 211148A02R00

Cost : 810 US Dollars

Availability : 99 calendar days from receipt of order

The Kit availability given above is the standard lead time from the date of your purchase order. If you require the Kit(s) before this time, please include a retrofit planning schedule with your order so that we can try to comply with your requirements.

The sales terms (costs and availability) are estimated in relation to economic conditions at the issue date of the Service Bulletin.

Kit 211148A03R00

Cost : 810 US Dollars

Availability : 99 calendar days from receipt of order

The Kit availability given above is the standard lead time from the date of your purchase order. If you require the Kit(s) before this



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time, please include a retrofit planning schedule with your order so that we can try to comply with your requirements.

The sales terms (costs and availability) are estimated in relation to economic conditions at the issue date of the Service Bulletin.

Kit 211148A04R00

Cost : 810 US Dollars

Availability : 99 calendar days from receipt of order

The Kit availability given above is the standard lead time from the date of your purchase order. If you require the Kit(s) before this time, please include a retrofit planning schedule with your order so that we can try to comply with your requirements.

The sales terms (costs and availability) are estimated in relation to economic conditions at the issue date of the Service Bulletin.

Kit 211148A05R00

Cost : 810 US Dollars

Availability : 99 calendar days from receipt of order

The Kit availability given above is the standard lead time from the date of your purchase order. If you require the Kit(s) before this time, please include a retrofit planning schedule with your order so that we can try to comply with your requirements.

The sales terms (costs and availability) are estimated in relation to economic conditions at the issue date of the Service Bulletin.

Kit 211148A06R00

Cost : 810 US Dollars

Availability : 99 calendar days from receipt of order

The Kit availability given above is the standard lead time from the date of your purchase order. If you require the Kit(s) before this time, please include a retrofit planning schedule with your order so that we can try to comply with your requirements.

The sales terms (costs and availability) are estimated in relation to economic conditions at the issue date of the Service Bulletin.

Kit 211148A07R00

Cost : 810 US Dollars

Availability : 99 calendar days from receipt of order

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The Kit availability given above is the standard lead time from the date of your purchase order. If you require the Kit(s) before this time, please include a retrofit planning schedule with your order so that we can try to comply with your requirements.

The sales terms (costs and availability) are estimated in relation to economic conditions at the issue date of the Service Bulletin.

Kit 211148A08R00

Cost : 810 US Dollars

Availability : 99 calendar days from receipt of order

The Kit availability given above is the standard lead time from the date of your purchase order. If you require the Kit(s) before this time, please include a retrofit planning schedule with your order so that we can try to comply with your requirements.

The sales terms (costs and availability) are estimated in relation to economic conditions at the issue date of the Service Bulletin.

Kit 211148A09R00

Cost : 810 US Dollars

Availability : 99 calendar days from receipt of order

The Kit availability given above is the standard lead time from the date of your purchase order. If you require the Kit(s) before this time, please include a retrofit planning schedule with your order so that we can try to comply with your requirements.

The sales terms (costs and availability) are estimated in relation to economic conditions at the issue date of the Service Bulletin.

Kit 211148A10R00

Cost : 810 US Dollars

Availability : 99 calendar days from receipt of order

The Kit availability given above is the standard lead time from the date of your purchase order. If you require the Kit(s) before this time, please include a retrofit planning schedule with your order so that we can try to comply with your requirements.

The sales terms (costs and availability) are estimated in relation to economic conditions at the issue date of the Service Bulletin.

Kit 211148A11R00

Cost : 810 US Dollars

Availability : 99 calendar days from receipt of order



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The Kit availability given above is the standard lead time from the date of your purchase order. If you require the Kit(s) before this time, please include a retrofit planning schedule with your order so that we can try to comply with your requirements.

The sales terms (costs and availability) are estimated in relation to economic conditions at the issue date of the Service Bulletin.

B. INDUSTRY SUPPORT INFORMATION

AIRBUS will not provide industrial support for accomplishment of this Service Bulletin.

C. LIST OF COMPONENTS

Kit 211148A01R01

ITEM	NEW PART No.	QTY	UM	KEYWORD	ITEM	OLD PART No.	INT	INST	DISP
2	NSA8205-171	2		Packing					
3	NAS1153E8	20		Screw					
4	D5311044620096	2		Plug					
8	E0718-10	1	M	Tubing					
10	ABS0772A02A	3		Clip					
18	NSA938005-14	2		Cap					
19	NSA938005-12	1		Cap					
20	NSA938005-10	1		Cap					
	EN3155-016M2222	6		Contact					
	NSA935401-04	62		Tie					
	NSA935401-06	50		Tie					
	NSA936604CA0931	4		Cap					

NOTE : Refer to the AIRBUS Standards Manual (SM) if you find part numbers of hardware components in the related kit(s) which you cannot identify in the LIST OF COMPONENTS of this Service Bulletin. The SM will give you the correct part number to part number relationship.

Kit 211148A02R00

ITEM	NEW PART No.	QTY	UM	KEYWORD	ITEM	OLD PART No.	INT	INST	DISP
12	D33110183A00	1		Plate	(12)	D33110123B00	03	*	
	ASNA2083BG2-10	4		Screw					

NOTE : For definitions of interchangeability codes shown in column INT. refer to ATA Common Support Data Dictionary (CSDD), chapter 2.

NOTE (*) Discard

Kit 211148A03R00

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ITEM	NEW PART No.	QTY	UM	KEYWORD	ITEM	OLD PART No.	INT	INST	DISP
12	D33110046A00	1		Plate	(12)	D33110104A00	03	*	
	ASNA2083BG2-10	4		Screw					

NOTE : For definitions of interchangeability codes shown in column INT.
refer to ATA Common Support Data Dictionary (CSDD), chapter 2.

NOTE (*) Discard

NOTE : Refer to the AIRBUS Standards Manual (SM) if you find part numbers of hardware components in the related kit(s) which you cannot identify in the LIST OF COMPONENTS of this Service Bulletin. The SM will give you the correct part number to part number relationship.

Kit 211148A04R00

ITEM	NEW PART No.	QTY	UM	KEYWORD	ITEM	OLD PART No.	INT	INST	DISP
12	D33110229A00	1		Plate	(12)	D33110110A00	03	*	
	ASNA2083BG2-10	4		Screw					

NOTE : For definitions of interchangeability codes shown in column INT.
refer to ATA Common Support Data Dictionary (CSDD), chapter 2.

NOTE (*) Discard

NOTE : Refer to the AIRBUS Standards Manual (SM) if you find part numbers of hardware components in the related kit(s) which you cannot identify in the LIST OF COMPONENTS of this Service Bulletin. The SM will give you the correct part number to part number relationship.

Kit 211148A05R00

ITEM	NEW PART No.	QTY	UM	KEYWORD	ITEM	OLD PART No.	INT	INST	DISP
12	D33110284A00	1		Plate	(12)	D33110086B00	03	*	
	ASNA2083BG2-10	4		Screw					

NOTE : For definitions of interchangeability codes shown in column INT.
refer to ATA Common Support Data Dictionary (CSDD), chapter 2.

NOTE (*) Discard

NOTE : Refer to the AIRBUS Standards Manual (SM) if you find part numbers of hardware components in the related kit(s) which you cannot identify in the LIST OF COMPONENTS of this Service Bulletin. The SM will give you the correct part number to part number relationship.

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Kit 211148A06R00

ITEM	NEW PART No.	QTY	UM	KEYWORD	ITEM	OLD PART No.	INT	INST	DISP
12	D33110285A00	1		Plate	(12)	D33110137A00	03	*	
	ASNA2083BG2-10	4		Screw					

NOTE : For definitions of interchangeability codes shown in column INT.
refer to ATA Common Support Data Dictionary (CSDD), chapter 2.

NOTE (*) Discard

NOTE : Refer to the AIRBUS Standards Manual (SM) if you find part numbers of hardware components in the related kit(s) which you cannot identify in the LIST OF COMPONENTS of this Service Bulletin. The SM will give you the correct part number to part number relationship.

Kit 211148A07R00

ITEM	NEW PART No.	QTY	UM	KEYWORD	ITEM	OLD PART No.	INT	INST	DISP
12	D33110286A00	1		Plate	(12)	D33110197A00	03	*	
	ASNA2083BG2-10	4		Screw					

NOTE : For definitions of interchangeability codes shown in column INT.
refer to ATA Common Support Data Dictionary (CSDD), chapter 2.

NOTE (*) Discard

NOTE : Refer to the AIRBUS Standards Manual (SM) if you find part numbers of hardware components in the related kit(s) which you cannot identify in the LIST OF COMPONENTS of this Service Bulletin. The SM will give you the correct part number to part number relationship.

Kit 211148A08R00

ITEM	NEW PART No.	QTY	UM	KEYWORD	ITEM	OLD PART No.	INT	INST	DISP
12	D33110255A00	1		Plate	(12)	D33110087A00	03	*	
	ASNA2083BG2-10	4		Screw					

NOTE : For definitions of interchangeability codes shown in column INT.
refer to ATA Common Support Data Dictionary (CSDD), chapter 2.

NOTE (*) Discard

NOTE : Refer to the AIRBUS Standards Manual (SM) if you find part numbers of hardware components in the related kit(s) which you cannot identify in the LIST OF COMPONENTS of this Service Bulletin. The SM will give you the correct part number to part number relationship.

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Kit 211148A09R00

ITEM	NEW PART No.	QTY	UM	KEYWORD	ITEM	OLD PART No.	INT	INST	DISP
12	D33110084A00	1		Plate	(12)	D33110141B00	03	*	
	ASNA2083BG2-10	4		Screw					

NOTE : For definitions of interchangeability codes shown in column INT.
refer to ATA Common Support Data Dictionary (CSDD), chapter 2.

NOTE (*) Discard

NOTE : Refer to the AIRBUS Standards Manual (SM) if you find part numbers of hardware components in the related kit(s) which you cannot identify in the LIST OF COMPONENTS of this Service Bulletin. The SM will give you the correct part number to part number relationship.

Kit 211148A10R00

ITEM	NEW PART No.	QTY	UM	KEYWORD	ITEM	OLD PART No.	INT	INST	DISP
12	D33110120A00	1		Plate	(12)	D33110271A00	03	*	
	ASNA2083BG2-10	4		Screw					

NOTE : For definitions of interchangeability codes shown in column INT.
refer to ATA Common Support Data Dictionary (CSDD), chapter 2.

NOTE (*) Discard

NOTE : Refer to the AIRBUS Standards Manual (SM) if you find part numbers of hardware components in the related kit(s) which you cannot identify in the LIST OF COMPONENTS of this Service Bulletin. The SM will give you the correct part number to part number relationship.

Kit 211148A11R00

ITEM	NEW PART No.	QTY	UM	KEYWORD	ITEM	OLD PART No.	INT	INST	DISP
12	D33110121A00	1		Plate	(12)	D00410458A00	03	*	
	ASNA2083BG2-10	4		Screw					

NOTE : For definitions of interchangeability codes shown in column INT.
refer to ATA Common Support Data Dictionary (CSDD), chapter 2.

NOTE (*) Discard

NOTE : Refer to the AIRBUS Standards Manual (SM) if you find part numbers of hardware components in the related kit(s) which you cannot identify in the LIST OF COMPONENTS of this Service Bulletin. The SM will give you the correct part number to part number relationship.

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D. LIST OF MATERIALS – OPERATOR SUPPLIED

(1) Consumable Materials

DESCRIPTION	REFERENCE TO CML MAT. No.	QTY PER A/C	INST DISP
Corrosion Inhibiting fillet Consistency	09-016	As required	
Chemical Conversion Coating Yellow Aluminum	13-002	As required	
Polyurethane Primer	16-001C	As required	
Lockwire MS20995C20	None	As required	

(2) Components

None

E. PARTS TO BE RE-IDENTIFIED BY THE OPERATOR

None

F. TOOLING – PRICE AND AVAILABILITY

None

G. SPECIAL TOOLS

None

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3. ACCOMPLISHMENT INSTRUCTIONS

A. GENERAL

WARNING : MAKE SURE THAT YOU OBEY ALL THE WARNINGS AND ALL THE CAUTIONS INCLUDED IN THE REFERENCED PROCEDURES.

CAUTION : ALWAYS OBEY THE PRECAUTIONS THAT FOLLOW TO KEEP ELECTRICAL WIRING IN A SATISFACTORY CONDITION (ELECTRICALLY AND MECHANICALLY SERVICEABLE). WHEN YOU DO MAINTENANCE WORK, REPAIRS OR MODIFICATIONS, ALWAYS KEEP ELECTRICAL WIRING, COMPONENTS AND THE WORK AREA AS CLEAN AS POSSIBLE, TO DO THIS :

- PUT PROTECTION, SUCH AS PLASTIC SHEETING, CLOTHS, ETC;
AS NECESSARY ON WIRING AND COMPONENTS.
- REGULARLY REMOVE ALL SHAVINGS, UNWANTED MATERIAL AND OTHER CONTAMINATION.

THESE PRECAUTIONS WILL DECREASE THE RISK OF CONTAMINATION AND DAMAGE TO THE ELECTRICAL WIRING INSTALLATION.

IF THERE IS CONTAMINATION REFER TO ESPM 20-55-00.

(1) Preparation

- (a) Make sure that the aircraft is electrically grounded (Refer to AMM 12-34-24 Page block 201).
- (b) Put the access platforms in position.
- (c) Open the access doors 812, 822 and 824 (Refer to AMM 52-41-00 Page block 001).
- (d) Do the preparation procedure as specified in the removal/installation of the System Data Acquisition Concentrators (SDACs) (1WV1 and 1WV2) (Refer to AMM 31-55-34 Page block 401).
- (e) For the wiring parity of the SDACs, before any disconnection, note the figure number of the Hook-up Chart on which lines 1 and 3 corresponds to the existing aircraft wiring configuration (Refer to figure 20 or figure 21).
- (f) Make sure that the external connector(s) is (are) not connected to the aircraft receptacles EXT PW.
- (g) Make sure that there is (are) warning notice(s) on external power receptacle to tell persons not to connect an external power source.
- (h) Make sure that the EMER EXIT LT pushbutton switch on 25VU is on OFF position.

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- (i) In the cockpit, on the overhead panel 35VU release BAT pushbutton switches.
 - (j) Put a warning notice on panel 35VU to tell persons no to operate the batteries.
 - (k) Put a warning notice on refuel/defuel panel 800VU to tell persons no to operate the NORMAL BATTERY switch.
 - (l) Put a warning notice in the cockpit to tell persons not to start the APU.
 - (m) Put a warning notice in the cockpit to tell persons not to start the ENG 1, 2.
 - (n) Disconnection of the battery :
 - For job set-up, refer to removal/installation of the batteries (2PB1, 2PB2) (Refer to AMM 24-38-51 Page block 401).
 - Turn the knurled nut by a quarter turn.
 - Disconnect the batteries connectors.
 - Put blanking caps on the disconnected electrical.
- (2) Standard Practices
- (a) For the specification of the material numbers (Mat. No.), refer to the CML.
 - (b) Remove and install the fasteners (Refer to SRM 51-40-00).
 - (c) Drill and deburr the holes metallic structure (Refer to SRM 51-44-11).
 - (d) Countersink the holes, (Refer to SRM 51-46-11).
 - (e) Protect the part to be drilled with Chemical Conversion Coating Yellow Aluminum (Mat No. 13-002) and Polyurethane Primer (Mat No. 16-001C) .
 - (f) Torque the standard threaded fasteners (Refer to AMM 20-21-11 Page block 201).
 - (g) Apply Corrosion Inhibiting fillet Consistency (Mat No. 09-016)
 - (h) Renew the protection finish with Polyurethane Primer (Mat No. 16-001C) .
 - (i) For the alternate and substitute fasteners, (Refer to SRM 51-43-12).



(j) Obey the instructions for the general wiring installation, (Refer to ESPM 20-33-00).

(k) Do a continuity test of the modified wires.

B. MODIFICATION

(1) For Kit 211148A01R01

(a) In the avionics compartment, LH side, modify the inlet skin air valve (11HD).

Refer to figure 1 Sheet 2

Refer to figure 2 Sheets 1 and 2

Refer to figure 3

Refer to figure 4 Sheets 1 and 2

Refer to AMM 21-27-52 Page block 401

1 Remove :

Refer to AMM 21-27-52 Page block 401

Refer to figure 2 Sheet 1

FIN 11HD

10	Screw	Item (3)	Discard
1	Skin Air Valve	Item (5)	
1	Packing	Item (2)	Discard

NOTE : Temporarily retain the inlet skin air valve (11HD) item (5). This skin air valve is necessary to drill the holes of the new plug.

2 On bench :

Refer to figure 3

a Put in position on the inlet skin air valve (11HD) item (5) :

1 Plug D5311044620096 Item 4

b Make a mark for the drilling of the ten holes on the plug item 4.

c Remove the skin air inlet valve (11HD) item (5) and discard it.

d Drill the plug item 4 to match the ten holes, (Refer to figure 3).

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- e Countersink the ten holes of the plug item 4, (Refer to SRM 51-46-11.)

3 Modify the routing.

Refer to figure 4 Sheets 1 and 2

- a Disconnect the wires from the connector 11HD-A.

- b Remove :

FIN 11HD-A

- | | | | |
|---|-----------|----------|--------|
| 1 | Connector | Item (7) | Retain |
|---|-----------|----------|--------|

- c Pull the wires through the pressure seal 1303VP, out of the cover.

- d Connect the wires on the connector 11HD-A item (7) as per AWL.

- e Install on the connector as shown in view C.

Refer to figure 4 Sheet 2

- | | | | |
|-----|-------------|----------|--------|
| 1 M | Tubing-Heat | E0718-10 | Item 8 |
|-----|-------------|----------|--------|

attach with :

- | | | |
|---|-----------|--------------|
| 2 | Tie-Cable | NSA935401-04 |
|---|-----------|--------------|

- f Install the wires with the connector 11HD-A in the stowed position on the route 1M.

- g Attach with :

- | | | |
|----|-----------|--------------|
| 10 | Tie-Cable | NSA935401-04 |
|----|-----------|--------------|

or

- | | | |
|----|-----------|--------------|
| 10 | Tie-Cable | NSA935401-06 |
|----|-----------|--------------|

4 Install :

Refer to figure 2 Sheet 2

- | | | | |
|---|---------|----------------|--------|
| 1 | Packing | NSA8205-171 | Item 2 |
| 1 | Plug | D5311044620096 | Item 4 |

attach with :

- | | | | |
|----|-------|-----------|--------|
| 10 | Screw | NAS1153E8 | Item 3 |
|----|-------|-----------|--------|

- (b) In the avionics compartment, RH side, modify the outlet skin air valve (10HD).

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Refer to figure 1 Sheet 2

Refer to figure 3

Refer to figure 5 Sheets 1 and 2

Refer to figure 6 Sheets 1 and 2

Refer to AMM 21-27-53 Page block 401

1 Remove :

Refer to AMM 21-27-53 Page block 401

Refer to figure 5 Sheet 1

FIN 10HD

10	Screw	Item (3)	Discard
1	Skin Air Valve	Item (1)	
1	Packing	Item (2)	Discard

NOTE : Temporarily retain the outlet skin air valve (10HD).
This outlet skin air valve is necessary to drill the
holes of the new plug.

2 On bench :

Refer to figure 3

a Put in position on the outlet skin air valve (10HD) item
(1) :

1 Plug D5311044620096 Item 4

b Make a mark for the drilling of the ten holes on the
plug item 4.

c Remove the outlet skin air valve (10HD) item (1) and
discard it.

d Drill the plug item 4 to match the ten holes, (Refer to
figure 3).

e Countersink the ten holes of the plug item 4, (Refer to
SRM 51-46-11.)

3 Modify the routing.

Refer to figure 6 Sheets 1 and 2

a Disconnect the wires from the connector 10HD-A item (6).

b Remove :



FIN 10HD-A

- 1 Connector Item (6) Retain
 - c Pull the wires through the pressure seal 1310VP, out of the cover.
 - d Connect the wires on the connector 10HD-A item (6) as per AWL.
 - e Install on the connector as shown in view C.

Refer to figure 6 Sheet 2

Tubing-Heat E0718-10 Item 8
attach with :

- 2 Tie-Cable NSA935401-04
 - f Install the wires with the connector 10HD-A in the stowed position on the route 1M.
 - g Attach with :

10 Tie-Cable NSA935401-04
or

10 Tie-Cable NSA935401-06

4 Install :

Refer to figure 5 Sheet 2

- | | | | |
|---|---------|----------------|--------|
| 1 | Packing | NSA8205-171 | Item 2 |
| 1 | Plug | D5311044620096 | Item 4 |

attach with :

10 Screw NAS1153E8 Item 3

- (c) Modify the connectors of the fan-GND cool (8HD), the cooling unit-GND (12HD) and press SW-GND cool fan (13HD), in the avionics compartment.

Refer to figure 7 Sheets 1 and 2

- 1 Disconnect the connectors 8HD-A, 12HD-A, 12HD-B and 13HD-A.
- 2 Install on the connectors 8HD-A, 12HD-A, 12HD-B and 13HD-A as shown in view C.

Refer to figure 7 Sheet 2

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Tubing-Heat E0718-10 Item 8

Attach with :

8 Tie-Cable NSA935401-04

3 Install the wires with the connectors 8HD-A, 12HD-A, 12HD-B and 13HD-A in the stowed position

4 Attach with :

30 Tie-Cable NSA935401-04

or

30 Tie-Cable NSA935401-06

5 On the connector 8HD-A and 12HD-A, install :

Refer to figure 7 Sheet 2

2 Cap NSA938005-14 Item 18

6 On the connector 12HD-B, install :

Refer to figure 7 Sheet 2

1 Cap NSA938005-12 Item 19

7 On the connector 13HD-A, install :

Refer to figure 7 Sheet 2

1 Cap NSA938005-10 Item 20

(d) Modify the wiring on shelf 88VU, in the avionics compartment.

Refer to figure 1 Sheet 1

Refer to figure 12

Refer to figure 18

1 Modify the connection of the wires shown on the lines 1 thru 8 (Refer to figure 18).

use :

4 Cap NSA936604CA0931

(e) Modify the pin programming and the parity of the SDAC on the shelf 85VU and the shelf 86VU, in the avionics compartment.

Refer to figure 1 Sheet 1

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Refer to figure 13

Refer to figure 14

Refer to figure 15

Refer to figure 19

Refer to figure 20

Refer to figure 21

Refer to AMM 31-55-34

1 Remove the SDACs (1WV1 and 1WV2) (Refer to AMM 31-55-34 Page block 401).

2 Modify the pin programming.

Refer to figure 1 Sheet 1

Refer to figure 13

Refer to figure 19

a Modify the connection of wires shown on the lines 1 thru 4 (Refer to figure 19)

NOTE : If necessary, use :

2 Contact EN3155-016M2222

3 Modify the wiring parity of the SDACs.

Refer to figure 1 Sheet 1

Refer to figure 14

Refer to figure 15

Refer to figure 20

Refer to figure 21

a If during accomplishment of para. 3. A.(1) (e) you noted that the existing aircraft wiring correspond to lines 1 and 3 of figure 20 :

– Modify the connection of wires shown on the lines 1 thru 4 (Refer to figure 20).

NOTE : If necessary, use :

2 Contact EN3155-016M2222



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- b If during accomplishment of para. 3. A.(1) (e) you noted that the existing aircraft wiring correspond to lines 1 and 3 of figure 21 :

– Modify the connection of wires shown on the lines 1 thru 4 (Refer to figure 21).

NOTE : If necessary, use :

2 Contact EN3155-016M2222

4 Install the SDACs (1WV1 and 1WV2) (Refer to AMM 31-55-34 Page block 401).

- (f) Modify the equipment in the rear circuit breaker panel 122VU, in the cockpit.

Refer to figure 1 Sheet 1

Refer to figure 8

1 At the location W23, on the circuit breaker 3HD, install :

1 Clip-Locking ABS0772A02A Item 10

NOTE : Safety the clip on the circuit breaker with Lockwire MS20995C20 .

- (g) Modify the equipment in the power center-AC 123VU, in the cockpit.

Refer to figure 1 Sheet 1

Refer to figure 8

1 At the location AC11, on the circuit breaker 1HD, install :

1 Clip-Locking ABS0772A02A Item 10

NOTE : Safety the clamps on the circuit breaker with Lockwire MS20995C20 .

2 At the location AC03, on the circuit breaker 2HD, install :

1 Clip-Locking ABS0772A02A Item 10

NOTE : Safety the clamps on the circuit breaker with Lockwire MS20995C20 .

- (2) Kit 211148A02R00 , modify the equipment and the wiring in the overhead control panel 22VU, in the cockpit.

Refer to figure 1

Refer to figure 9 Sheets 1 and 2

Refer to figure 10

Refer to figure 16

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SERVICE BULLETIN

(a) Remove the overhead control panel 22VU.

(b) Remove :

Refer to figure 9 Sheet 1

FIN 9LF

1	Integrally Lighted Plate	D33110123B00	Item (12)	Discard
4	Screw			Discard

(c) Remove :

Refer to figure 9 Sheet 1 and 2

FIN 4HD-A

1	Connector		Item (13)	Discard
---	-----------	--	-----------	---------

FIN 4HD-1

1	Sleeve		Item (14)	Discard
---	--------	--	-----------	---------

FIN 4HD

1	Switch-Pushbutton		Item (16)	Discard
2	Screw		Item (15)	Discard

(d) Install :

Refer to figure 1

Refer to figure 9 Sheet 2

FIN 9LF

1	Integrally Lighted Plate	D33110183A00	Item 12	
---	-----------------------------	--------------	---------	--

Attach with :

4	Screw	ASNA2083BG2-10		
---	-------	----------------	--	--

(e) For Config. 01 , remove the wires shown on the lines 1 thru 17
(Refer to figure 16).

(f) For Config. 02 , remove the wires shown on the lines 1 thru 17
(Refer to figure 17).

(g) Install the overhead control panel 22VU.

(3) Kit 211148A03R00 , modify the equipment and the wiring in the
overhead control panel 22VU, in the cockpit.
Refer to figure 1

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Refer to figure 9 Sheets 1 and 2
Refer to figure 10
Refer to figure 16

(a) Remove the overhead control panel 22VU.

(b) Remove :

Refer to figure 9 Sheet 1

FIN 9LF

1	Integrally Lighted Plate	D33110104A00	Item (12)	Discard
4	Screw			Discard

(c) Remove :

FIN 4HD-A

1	Connector		Item (13)	Discard
---	-----------	--	-----------	---------

FIN 4HD-1

1	Sleeve		Item (14)	Discard
---	--------	--	-----------	---------

FIN 4HD

1	Switch-Pushbutton		Item (16)	Discard
2	Screw		Item (15)	Discard

(d) Install :

Refer to figure 1

Refer to figure 9 Sheet 2

FIN 9LF

1	Integrally Lighted Plate	D33110046A00	Item 12	
---	-----------------------------	--------------	---------	--

Attach with :

4	Screw	ASNA2083BG2-10		
---	-------	----------------	--	--

(e) For Config. 01 , remove the wires shown on the lines 1 thru 17
(Refer to figure 16).

(f) For Config. 02 , remove the wires shown on the lines 1 thru 17
(Refer to figure 17).

(g) Install the overhead control panel 22VU.



- (4) Kit 211148A04R00 , modify the equipment and the wiring in the overhead control panel 22VU, in the cockpit.

Refer to figure 1

Refer to figure 9 Sheets 1 and 2

Refer to figure 10

Refer to figure 16

- (a) Remove the overhead control panel 22VU.

- (b) Remove :

Refer to figure 9 Sheet 1

FIN 9LF

1	Integrally Lighted Plate	D33110110A00	Item (12)	Discard
4	Screw			Discard

- (c) Remove :

FIN 4HD-A

1	Connector		Item (13)	Discard
---	-----------	--	-----------	---------

FIN 4HD-1

1	Sleeve		Item (14)	Discard
---	--------	--	-----------	---------

FIN 4HD

1	Switch-Pushbutton		Item (16)	Discard
2	Screw		Item (15)	Discard

- (d) Install :

Refer to figure 9 Sheet 2

FIN 9LF

1	Integrally Lighted Plate	D33110229A00	Item 12	
---	-----------------------------	--------------	---------	--

Attach with :

4	Screw	ASNA2083BG2-10		
---	-------	----------------	--	--

- (e) For Config. 01 , remove the wires shown on the lines 1 thru 17 (Refer to figure 16).

- (f) For Config. 02 , remove the wires shown on the lines 1 thru 17 (Refer to figure 17).

- (g) Install the overhead control panel 22VU.



- (5) Kit 211148A05R00 , modify the equipment and the wiring in the overhead control panel 22VU, in the cockpit.

Refer to figure 1

Refer to figure 9 Sheets 1 and 2

Refer to figure 10

Refer to figure 16

- (a) Remove the overhead control panel 22VU.

- (b) Remove :

Refer to figure 9 Sheet 1

FIN 9LF

1	Integrally Lighted Plate	D33110086B00	Item (12)	Discard
4	Screw			Discard

- (c) Remove :

FIN 4HD-A

1	Connector		Item (13)	Discard
---	-----------	--	-----------	---------

FIN 4HD-1

1	Sleeve		Item (14)	Discard
---	--------	--	-----------	---------

FIN 4HD

1	Switch-Pushbutton		Item (16)	Discard
2	Screw		Item (15)	Discard

- (d) Install :

Refer to figure 9 Sheet 2

FIN 9LF

1	Integrally Lighted Plate	D33110284A00	Item 12	
---	-----------------------------	--------------	---------	--

Attach with :

4	Screw	ASNA2083BG2-10		
---	-------	----------------	--	--

- (e) For Config. 01 , remove the wires shown on the lines 1 thru 17 (Refer to figure 16).

- (f) For Config. 02 , remove the wires shown on the lines 1 thru 17 (Refer to figure 17).

- (g) Install the overhead control panel 22VU.



- (6) Kit 211148A06R00 , modify the equipment and the wiring in the overhead control panel 22VU, in the cockpit.

Refer to figure 1

Refer to figure 9 Sheets 1 and 2

Refer to figure 10

Refer to figure 16

- (a) Remove the overhead control panel 22VU.

- (b) Remove :

Refer to figure 9 Sheet 1

FIN 9LF

1	Integrally Lighted Plate	D33110137A00	Item (12)	Discard
4	Screw			Discard

- (c) Remove :

FIN 4HD-A

1	Connector		Item (13)	Discard
---	-----------	--	-----------	---------

FIN 4HD-1

1	Sleeve		Item (14)	Discard
---	--------	--	-----------	---------

FIN 4HD

1	Switch-Pushbutton		Item (16)	Discard
2	Screw		Item (15)	Discard

- (d) Install :

Refer to figure 9 Sheet 2

FIN 9LF

1	Integrally Lighted Plate	D33110285A00	Item 12	
---	-----------------------------	--------------	---------	--

Attach with :

4	Screw	ASNA2083BG2-10		
---	-------	----------------	--	--

- (e) For Config. 01 , remove the wires shown on the lines 1 thru 17 (Refer to figure 16).

- (f) For Config. 02 , remove the wires shown on the lines 1 thru 17 (Refer to figure 17).

- (g) Install the overhead control panel 22VU.



- (7) Kit 211148A07R00 , modify the equipment and the wiring in the overhead control panel 22VU, in the cockpit.

Refer to figure 1

Refer to figure 9 Sheets 1 and 2

Refer to figure 10

Refer to figure 16

- (a) Remove the overhead control panel 22VU.

- (b) Remove :

Refer to figure 9 Sheet 1

FIN 9LF

1	Integrally Lighted Plate	D33110197A00	Item (12)	Discard
4	Screw			Discard

- (c) Remove :

FIN 4HD-A

1	Connector		Item (13)	Discard
---	-----------	--	-----------	---------

FIN 4HD-1

1	Sleeve		Item (14)	Discard
---	--------	--	-----------	---------

FIN 4HD

1	Switch-Pushbutton		Item (16)	Discard
2	Screw		Item (15)	Discard

- (d) Install :

Refer to figure 9 Sheet 2

FIN 9LF

1	Integrally Lighted Plate	D33110286A00	Item 12	
---	-----------------------------	--------------	---------	--

Attach with :

4	Screw	ASNA2083BG2-10		
---	-------	----------------	--	--

- (e) For Config. 01 , remove the wires shown on the lines 1 thru 17 (Refer to figure 16).

- (f) For Config. 02 , remove the wires shown on the lines 1 thru 17 (Refer to figure 17).

- (g) Install the overhead control panel 22VU.



- (8) Kit 211148A08R00 , modify the equipment and the wiring in the overhead control panel 22VU, in the cockpit.

Refer to figure 1

Refer to figure 9 Sheets 1 and 2

Refer to figure 10

Refer to figure 16

- (a) Remove the overhead control panel 22VU.

- (b) Remove :

Refer to figure 9 Sheet 1

FIN 9LF

1	Integrally Lighted Plate	D33110087A00	Item (12)	Discard
4	Screw			Discard

- (c) Remove :

FIN 4HD-A

1	Connector		Item (13)	Discard
---	-----------	--	-----------	---------

FIN 4HD-1

1	Sleeve		Item (14)	Discard
---	--------	--	-----------	---------

FIN 4HD

1	Switch-Pushbutton		Item (16)	Discard
2	Screw		Item (15)	Discard

- (d) Install :

Refer to figure 9 Sheet 2

FIN 9LF

1	Integrally Lighted Plate	D33110255A00	Item 12	
---	-----------------------------	--------------	---------	--

Attach with :

4	Screw	ASNA2083BG2-10		
---	-------	----------------	--	--

- (e) For Config. 01 , remove the wires shown on the lines 1 thru 17 (Refer to figure 16).

- (f) For Config. 02 , remove the wires shown on the lines 1 thru 17 (Refer to figure 17).

- (g) Install the overhead control panel 22VU.



- (9) Kit 211148A09R00 , modify the equipment and the wiring in the overhead control panel 22VU, in the cockpit.

Refer to figure 1

Refer to figure 9 Sheets 1 and 2

Refer to figure 10

Refer to figure 16

- (a) Remove the overhead control panel 22VU.

- (b) Remove :

Refer to figure 9 Sheet 1

FIN 9LF

1	Integrally Lighted Plate	D33110141B00	Item (12)	Discard
4	Screw			Discard

- (c) Remove :

FIN 4HD-A

1	Connector		Item (13)	Discard
---	-----------	--	-----------	---------

FIN 4HD-1

1	Sleeve		Item (14)	Discard
---	--------	--	-----------	---------

FIN 4HD

1	Switch-Pushbutton		Item (16)	Discard
2	Screw		Item (15)	Discard

- (d) Install :

Refer to figure 9 Sheet 2

FIN 9LF

1	Integrally Lighted Plate	D33110084A00	Item 12	
---	-----------------------------	--------------	---------	--

Attach with :

4	Screw	ASNA2083BG2-10		
---	-------	----------------	--	--

- (e) For Config. 01 , remove the wires shown on the lines 1 thru 17 (Refer to figure 16).

- (f) For Config. 02 , remove the wires shown on the lines 1 thru 17 (Refer to figure 17).

- (g) Install the overhead control panel 22VU.



- (10) Kit 211148A10R00 , modify the equipment and the wiring in the overhead control panel 22VU, in the cockpit.

Refer to figure 1

Refer to figure 9 Sheets 1 and 2

Refer to figure 10

Refer to figure 16

- (a) Remove the overhead control panel 22VU.

- (b) Remove :

Refer to figure 9 Sheet 1

FIN 9LF

1	Integrally Lighted Plate	D33110271A00	Item (12)	Discard
4	Screw			Discard

- (c) Remove :

FIN 4HD-A

1	Connector		Item (13)	Discard
---	-----------	--	-----------	---------

FIN 4HD-1

1	Sleeve		Item (14)	Discard
---	--------	--	-----------	---------

FIN 4HD

1	Switch-Pushbutton		Item (16)	Discard
2	Screw		Item (15)	Discard

- (d) Install :

Refer to figure 9 Sheet 2

FIN 9LF

1	Integrally Lighted Plate	D33110120A00	Item 12	
---	-----------------------------	--------------	---------	--

Attach with :

4	Screw	ASNA2083BG2-10		
---	-------	----------------	--	--

- (e) For Config. 01 , remove the wires shown on the lines 1 thru 17 (Refer to figure 16).

- (f) For Config. 02 , remove the wires shown on the lines 1 thru 17 (Refer to figure 17).

- (g) Install the overhead control panel 22VU.



- (11) Kit 211148A11R00 , modify the equipment and the wiring in the overhead control panel 22VU, in the cockpit.

Refer to figure 1

Refer to figure 9 Sheets 1 and 2

Refer to figure 10

Refer to figure 16

- (a) Remove the overhead control panel 22VU.

- (b) Remove :

Refer to figure 9 Sheet 1

FIN 9LF

1	Integrally Lighted Plate	D00410458A00	Item (12)	Discard
4	Screw			Discard

- (c) Remove :

FIN 4HD-A

1	Connector		Item (13)	Discard
---	-----------	--	-----------	---------

FIN 4HD-1

1	Sleeve		Item (14)	Discard
---	--------	--	-----------	---------

FIN 4HD

1	Switch-Pushbutton		Item (16)	Discard
2	Screw		Item (15)	Discard

- (d) Install :

Refer to figure 9 Sheet 2

FIN 9LF

1	Integrally Lighted Plate	D33110121A00	Item 12	
---	-----------------------------	--------------	---------	--

Attach with :

4	Screw	ASNA2083BG2-10		
---	-------	----------------	--	--

- (e) For Config. 01 , remove the wires shown on the lines 1 thru 17 (Refer to figure 16).

- (f) For Config. 02 , remove the wires shown on the lines 1 thru 17 (Refer to figure 17).

- (g) Install the overhead control panel 22VU.



C. TEST

(1) Job set-up :

(a) Re-connect the battery :

- For the job set-up, refer to the removal/installation of the batteries (2PB1, 2PB2), (Refer to AMM 24-38-51 Page block 401),
- Remove the blanking cap from the electrical connectors,
- make sure that the electrical connectors are clean and in correct condition,
- connect the connector of the batteries,
- turn the knurled nut by a quarter turn,

(b) Remove all the warning notices.

(c) Restore the systems and the aircraft to the normal operating condition.

(2) Tests.

(a) Initial conditions :

- 1 On each connector which were disconnected, do a visual check, connector by connector as defined by ESPM 20-51-10, to make sure that :
 - the label of the plug is the same as the label on the receptacle,
 - the plug is correctly locked.

(b) Tests :

NOTE : If the aircraft is operated in CAT 3 conditions, you must also do this test : land CAT 3 capability test (Refer to AMM 22-97-00 Page block 501).

- 1 Do the operational test of the instrument and panel integral lighting (Refer to AMM 33-13-00 Page block 501).
- 2 Do the operational test of the annunciator light test system in the cockpit (Refer to AMM 33-14-00 Page block 501).
- 3 Do the operational test of the avionics equipment ventilation system (Refer to AMM 21-26-00 Page block 501).
- 4 Do the operational test of the Automatic Flight System (AFS) (Refer to AMM 22-96-00 Page block 501).

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- 5 Do the operational test of the Electronic Instrument System (EIS) (Refer to AMM 31-60-00 Page block 501).
 - 6 Do the operational test of the Electronic Centralized Aircraft Monitoring (ECAM) (Refer to AMM 31-50-00 Page block 501).
 - 7 Do the ground scanning of the central warning system (Refer to AMM 31-50-00 Page block 501).
- (3) Tests after disconnection/connection of the overhead control panel 22VU :
- (a) Do the operational check of override function (BLOWER AND EXTRACT), (Refer to AMM 21-26-00 Page block 501).
 - (b) Do the operational test of the cabin recirculation fans 14HG and 15HG, (Refer to AMM 21-21-00 Page block 501).
 - (c) Do the operational test of the avionics equipment ground cooling fault warning, (Refer to AMM 21-27-00 Page block 501).
 - (d) Do the operational test of the FWD cargo compartment heating system, (Refer to AMM 21-43-00 Page block 501).
 - (e) Do the operational test of the landscape camera, (Refer to AMM 23-26-00 Page block 501).
 - (f) Do the operational test of the cargo compartment smoke detection system by Push to Test (PTT), (Refer to AMM 26-16-00 Page block 501).
 - (g) Do the operational test of the cargo compartment smoke detection with the Centralized Fault Display System (CFDS) to confirm isolation valvelatching circuit integrity, (Refer to AMM 26-16-00 Page block 501).
 - (h) Do the operational test of the ground cooling isolation valve to verify closing in case of smoke warning, (Refer to AMM 26-16-00 Page block 501).
 - (i) Do the operational test of the cargo compartment fire extinguishing system, (Refer to AMM 26-23-00 Page block 501).
 - (j) Do the check of the firing circuit continuity and do the operational test of priority switching, (Refer to AMM 26-23-00 Page block 501).
 - (k) Do the functional test of time delay for illumination of "DISCH AGENT 2" reminder light, (Refer to AMM 26-23-00 Page block 501).
 - (l) Do the continuity check of the cargo compartment firing circuit, (Refer to AMM 26-23-00 Page block 501).



- (m) Do the inspection/check of the operation of the windshield wipers, (Refer to AMM 30-45-00 Page block 601).
- (n) Do the functional test of the windshield rain protection, (Refer to AMM 30-45-00 Page block 501).
- (o) Do the operational test of the engine electronic control, (Refer to AMM 73-22-34 Page block 501).

D. CLOSE UP

- (1) Do the close-up procedure as specified in the removal/installation of the SDACs (1WV1 and 1WV2) (Refer to AMM 31-55-34 Page block 401).
- (2) Close the access doors 812, 822 and 824 (Refer to AMM 52-41-00 Page block 001).
- (3) Remove the access platforms.

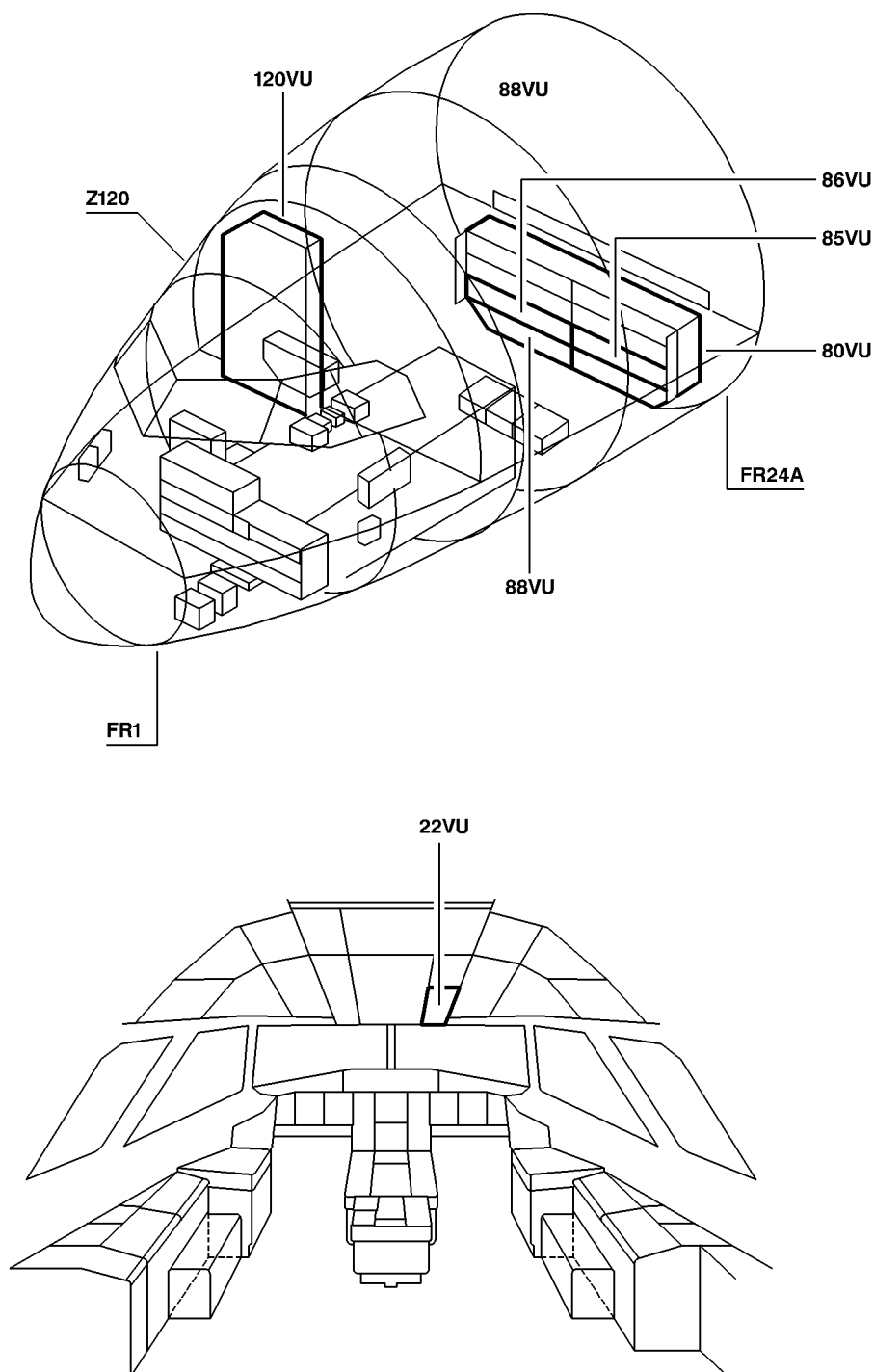
E. DOCUMENTATION

Write in the applicable aircraft records that you have done all the work given in the Service Bulletin.



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Figure 1 Sheet 1
Location of the Equipment

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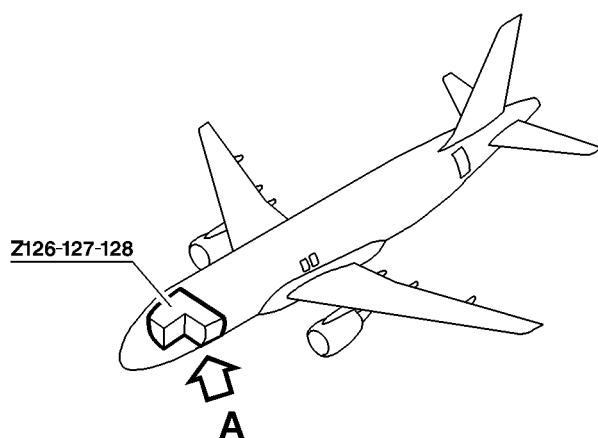
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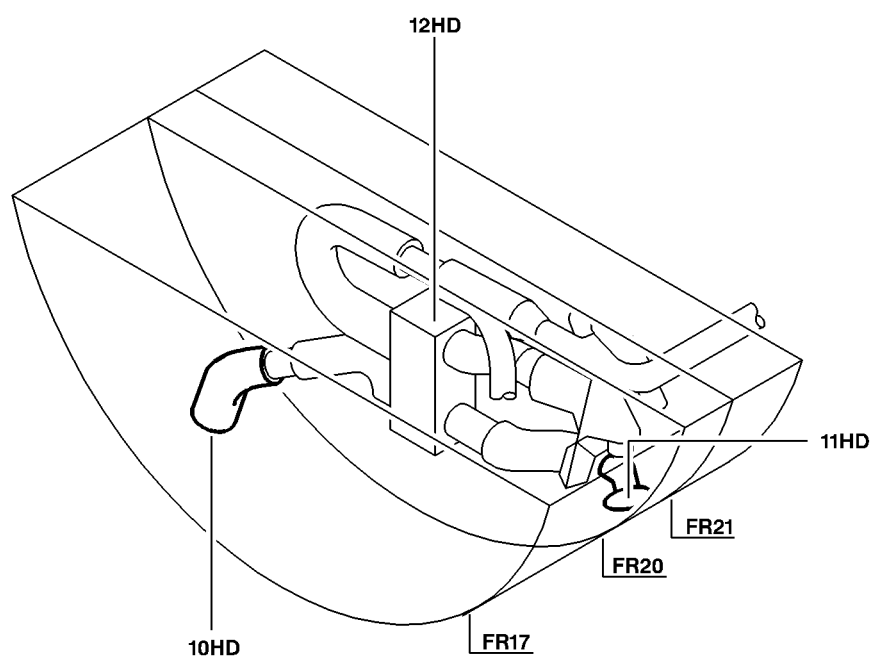


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SERVICE BULLETIN



A



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Figure 1 Sheet 2
Location of the Equipment

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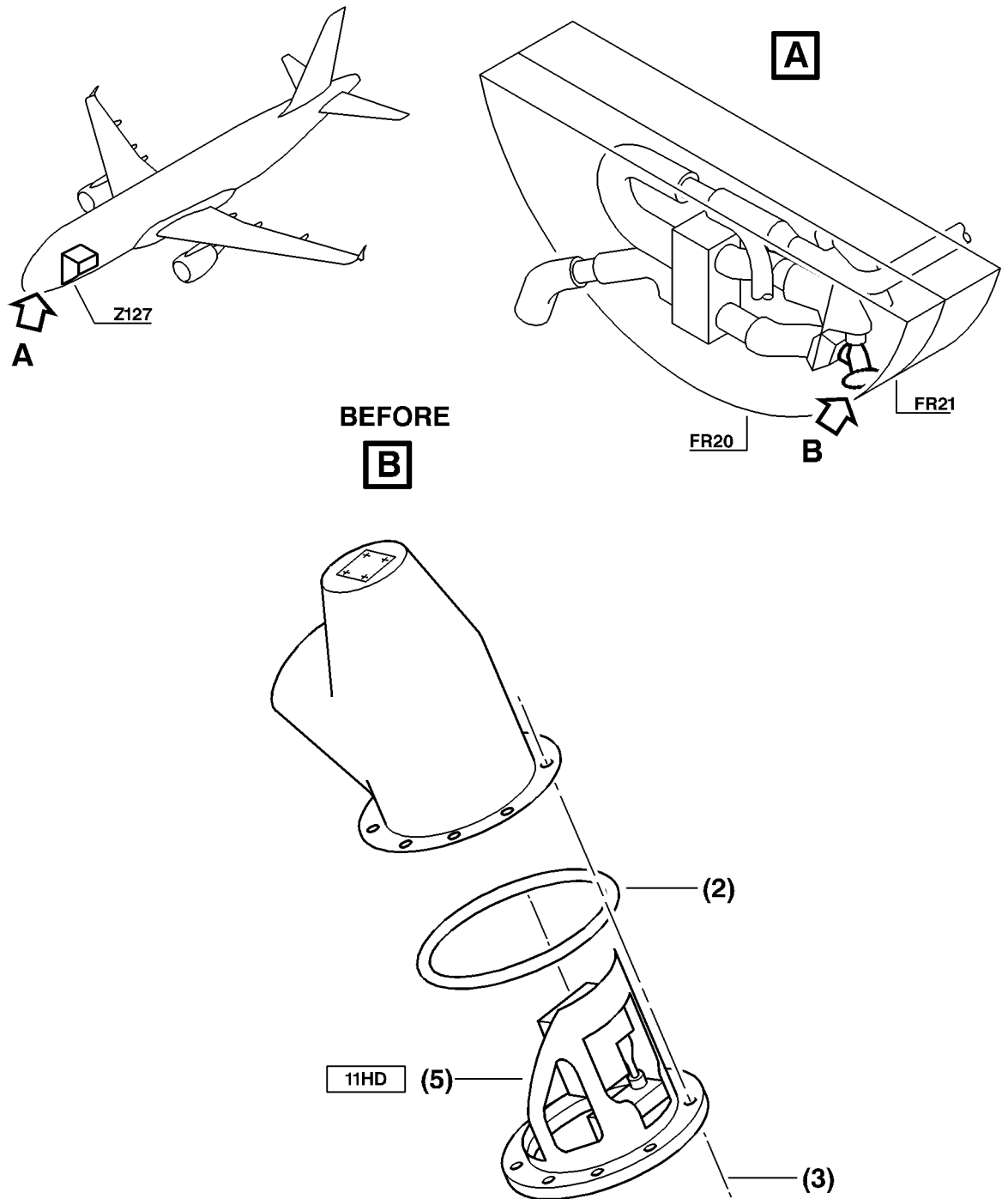
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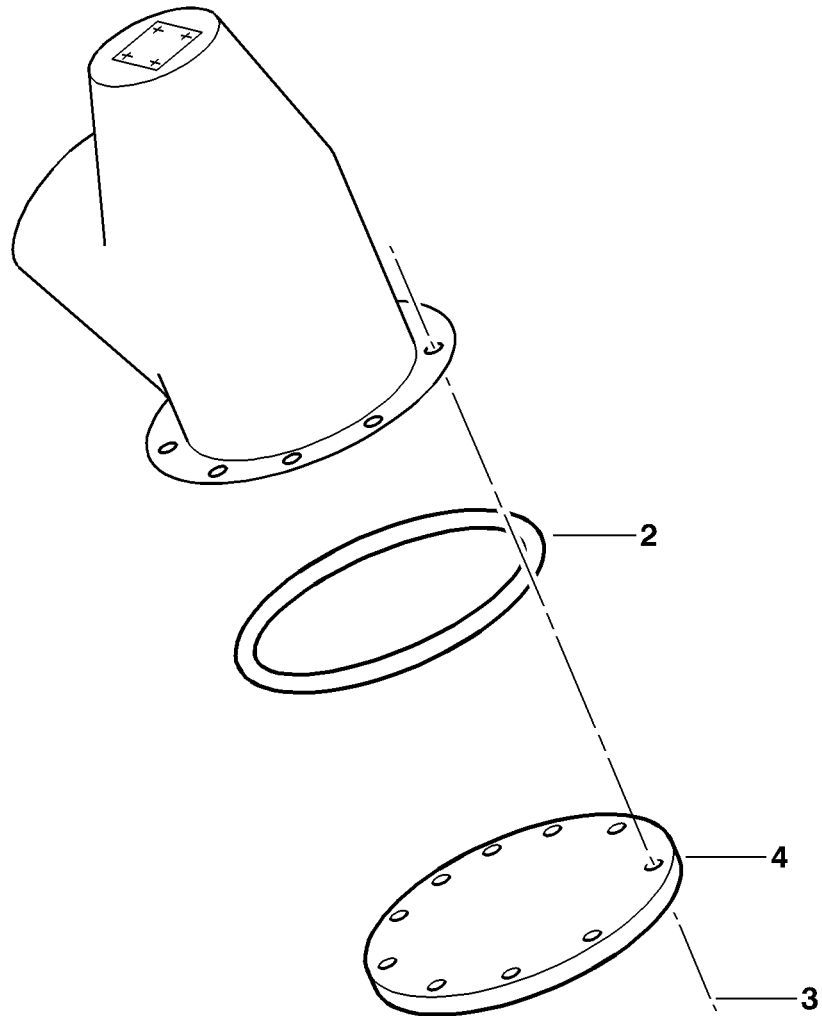
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AFTER

B



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Figure 2 Sheet 2
Modification of the Inlet Skin Air valve (11HD)

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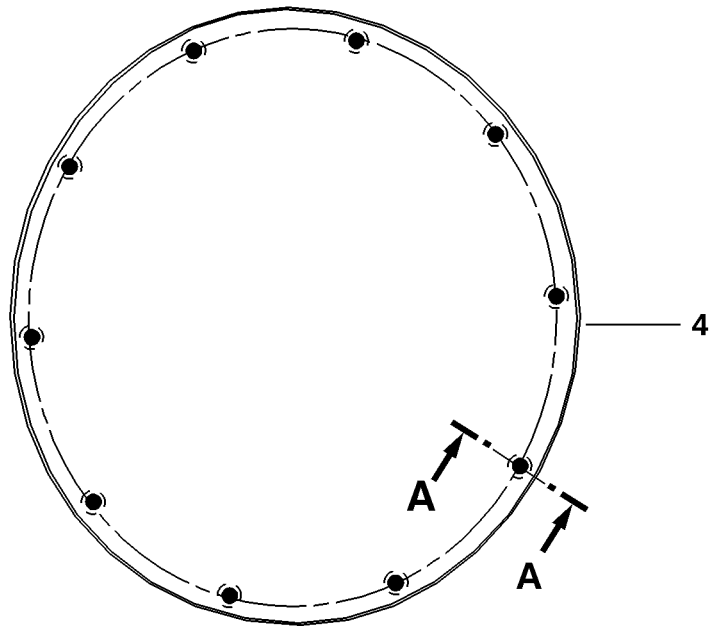
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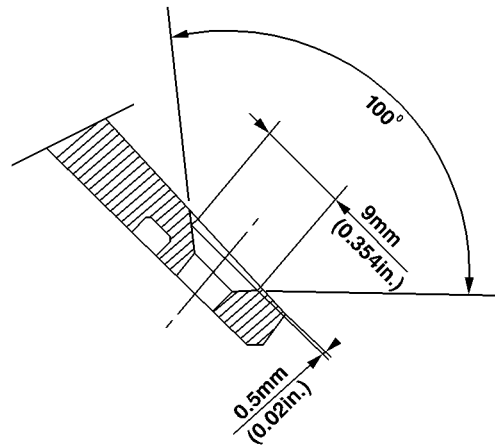
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ON BENCH



SECTION A-A



HOLE	NEW DRILLING HOLE DIA.			
	MAX.		MIN.	
	mm	in.	mm	in.
●	5.70	0.224	5.60	0,220

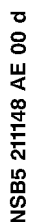
Figure 3 Sheet 1
Modification of the Plug on the Bench

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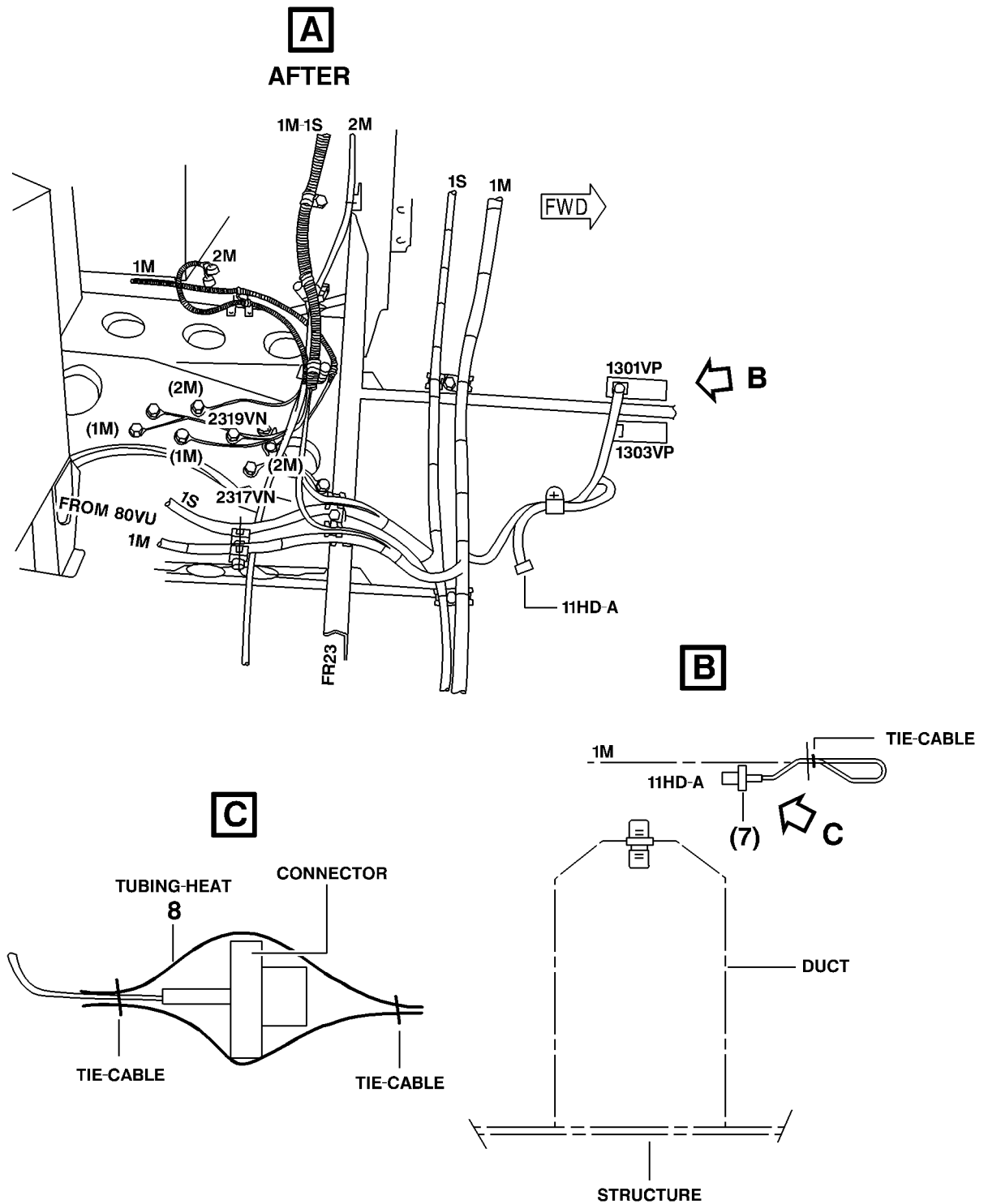


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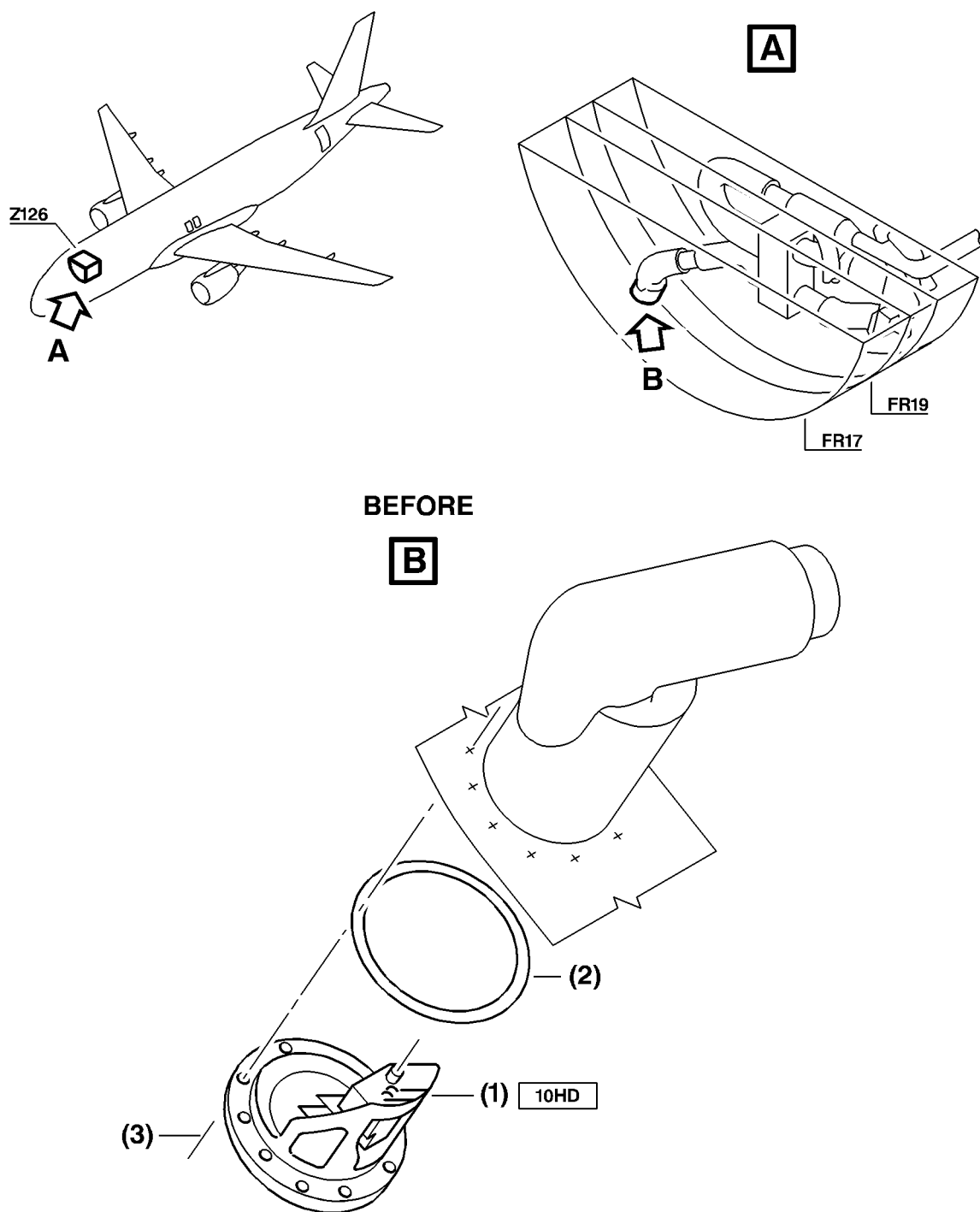
Figure 4 Sheet 2
Modification of the Routing in the LH FWD Avionics Compartment

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Figure 5 Sheet 1
Modification of the Outlet Skin Air Valve (10HD)

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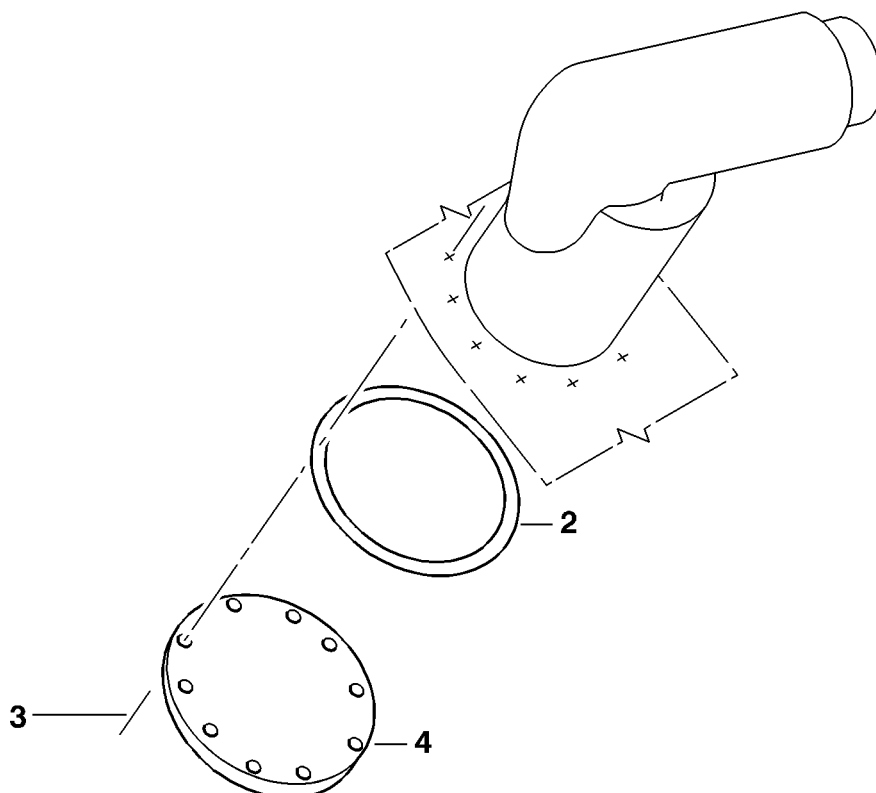
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AFTER

B



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Figure 5 Sheet 2
Modification of the Outlet Skin Air Valve (10HD)

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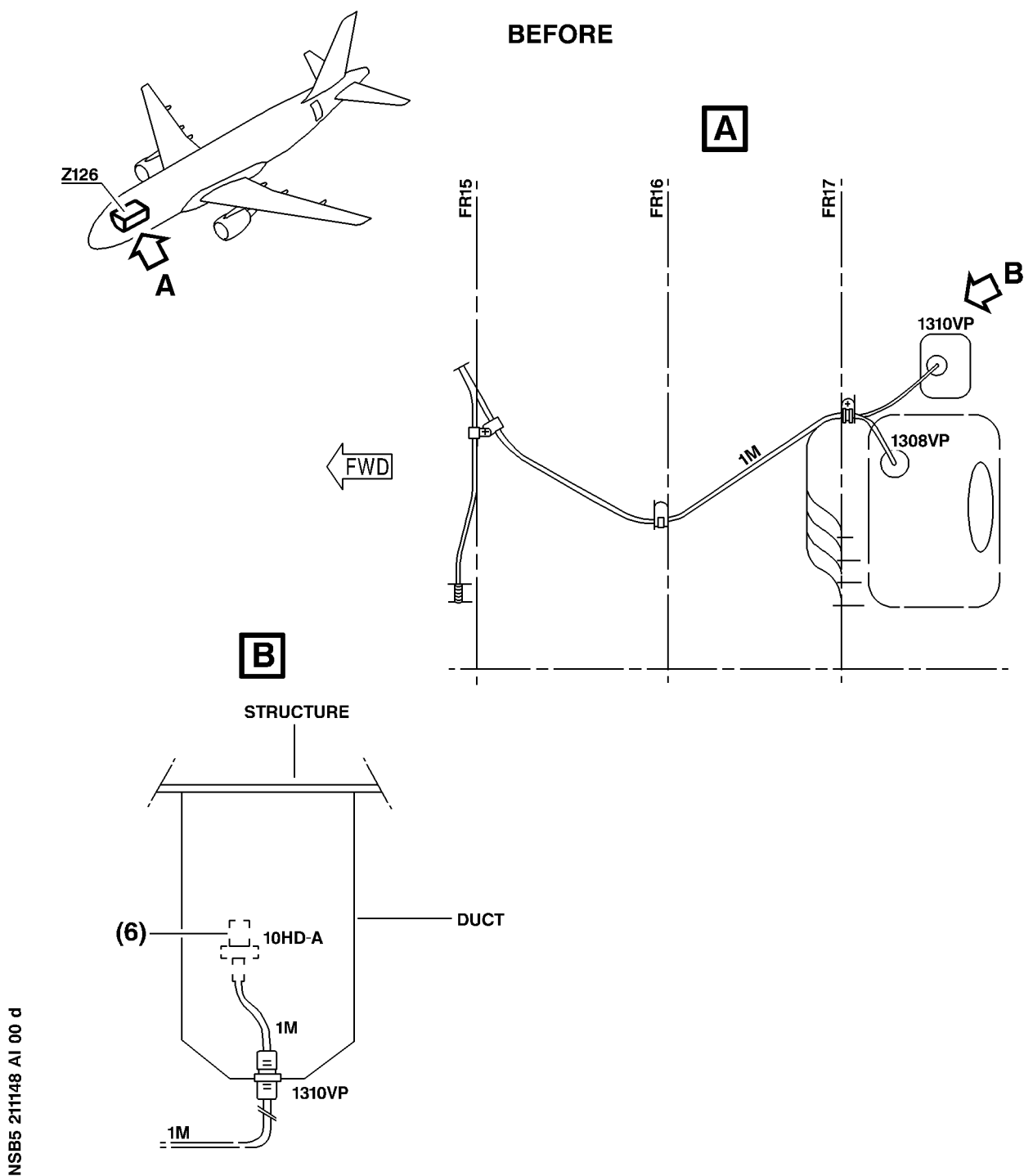


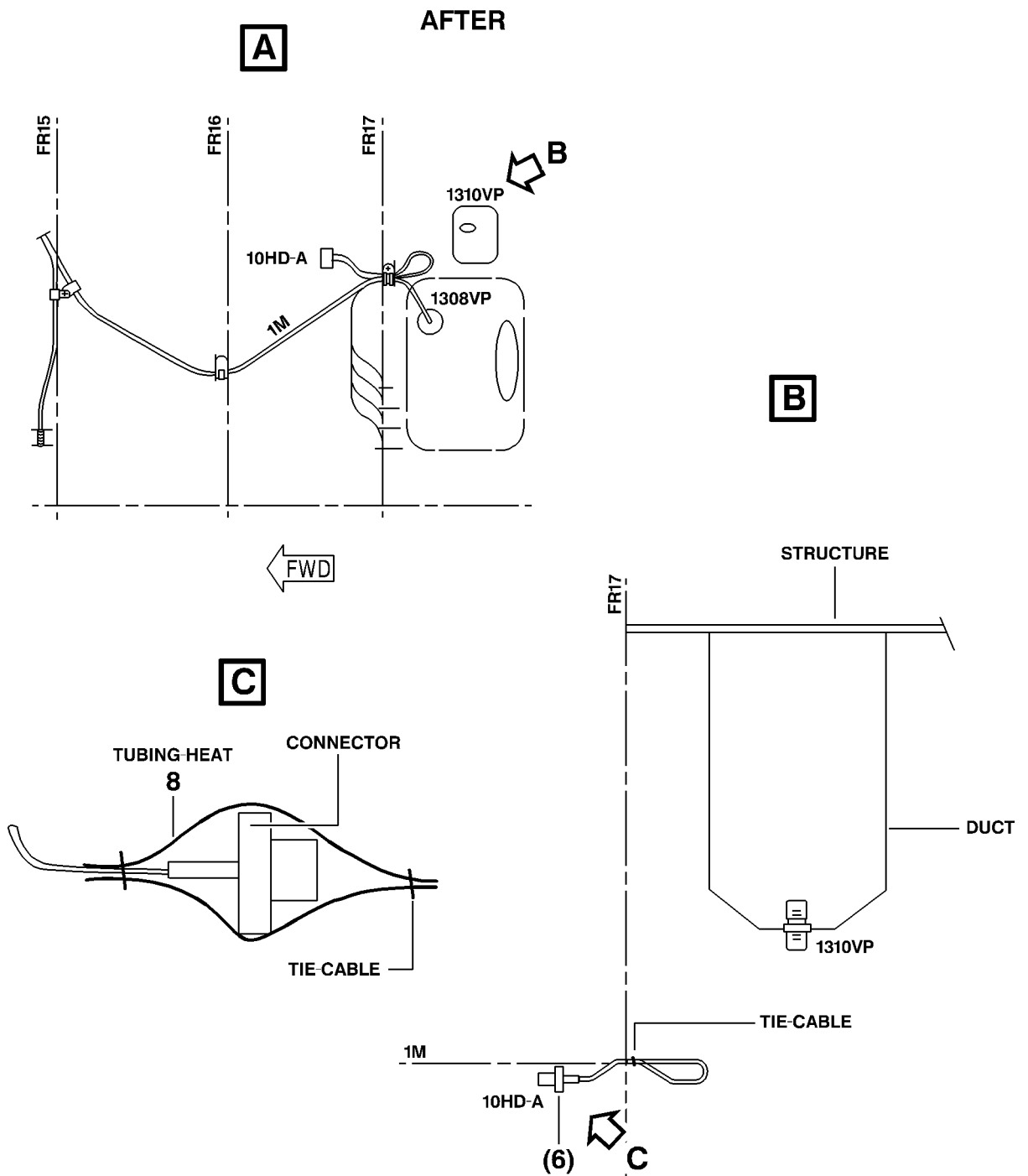
Figure 6 Sheet 1
Modification of the Routing in the RH Avionics Compartment

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Figure 6 Sheet 2
Modification of the Routing in the RH Avionics Compartment

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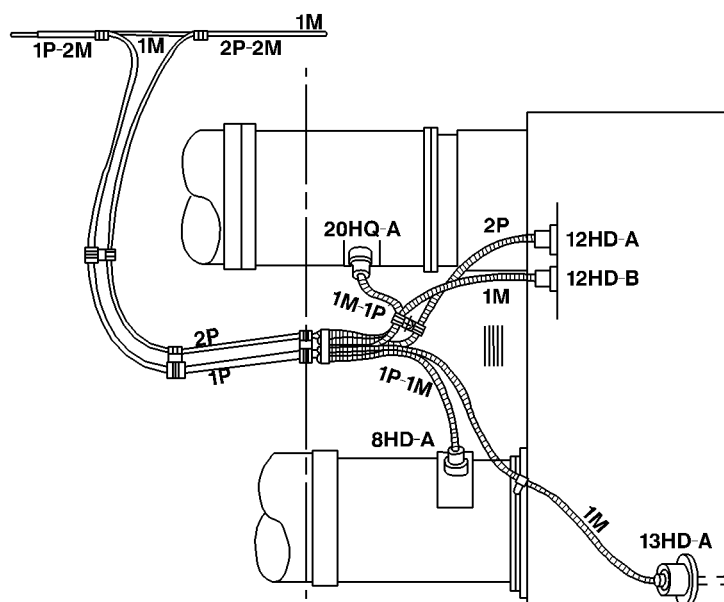
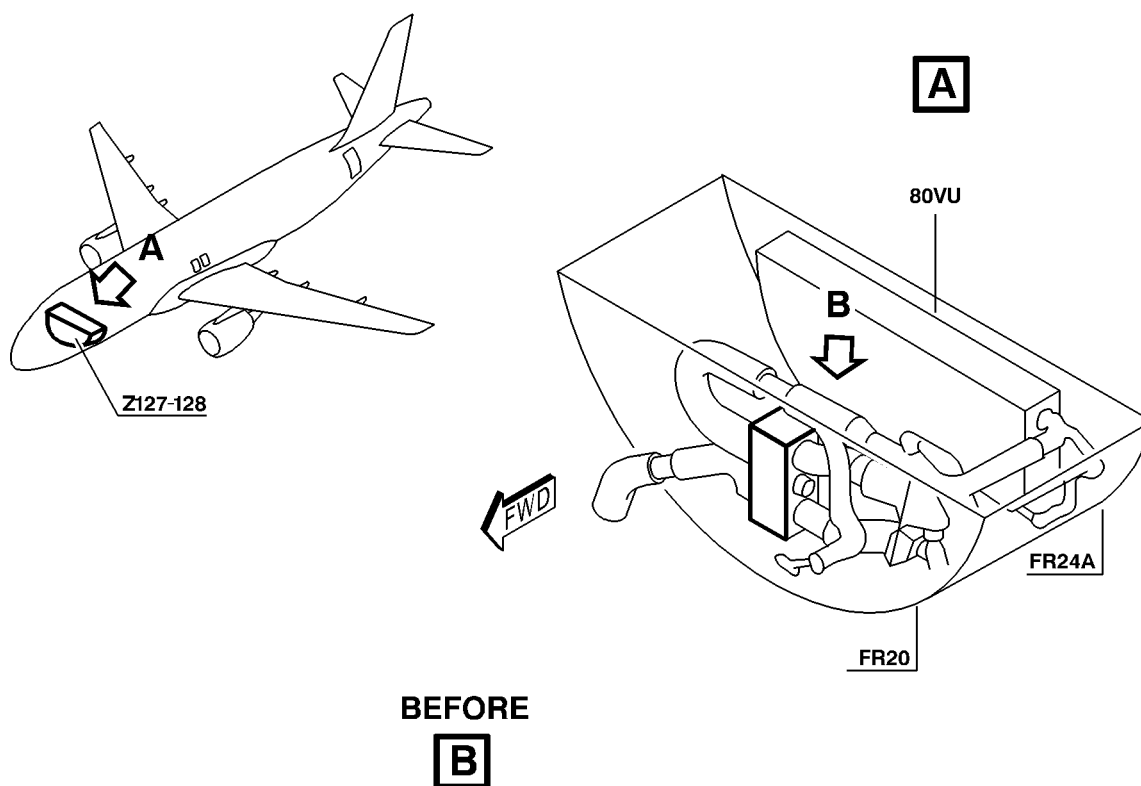


Figure 7 Sheet 1
Modification of the Connectors in the Avionics Compartment

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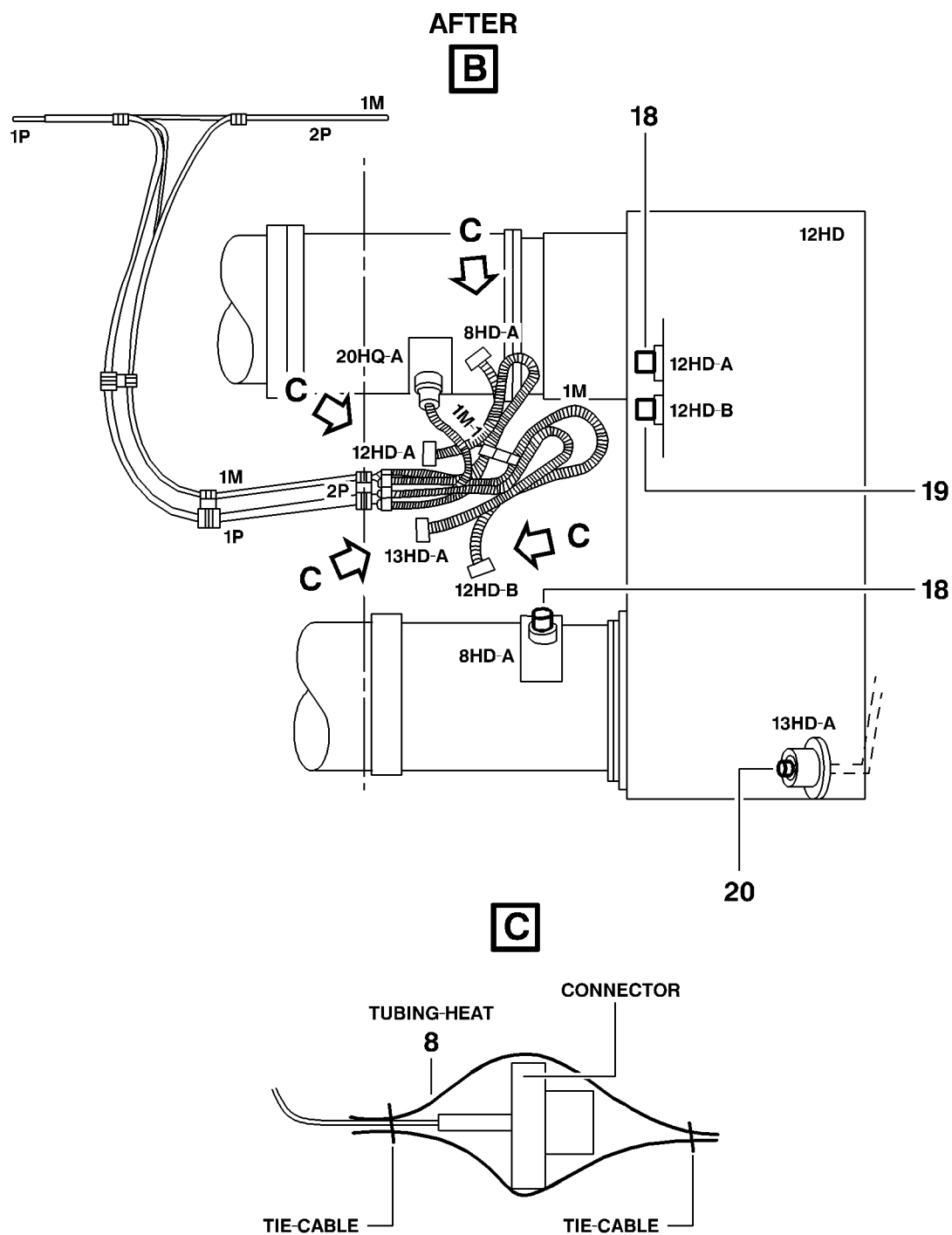
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Figure 7 Sheet 2
Modification of the Connectors in the Avionics Compartment

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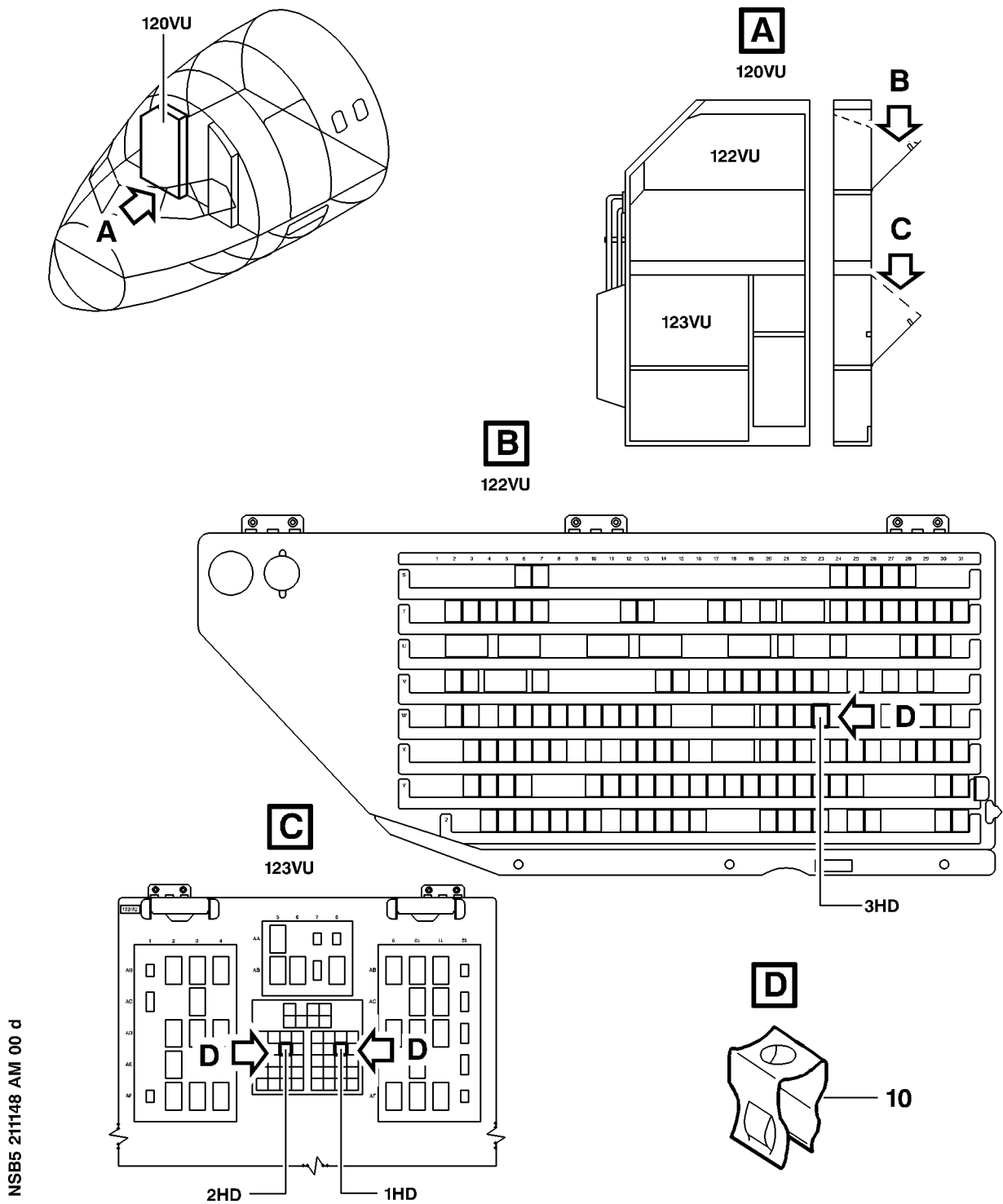
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Figure 8 Sheet 1
Modification of the Equipment in the Cockpit

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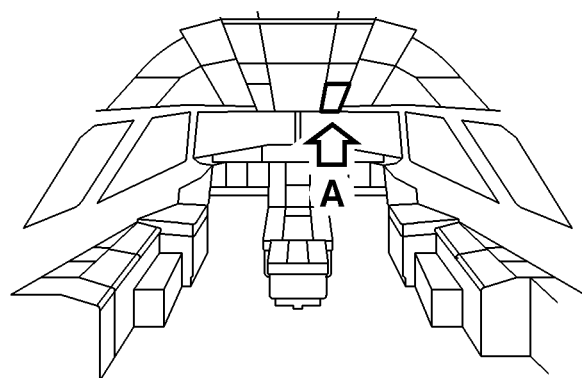
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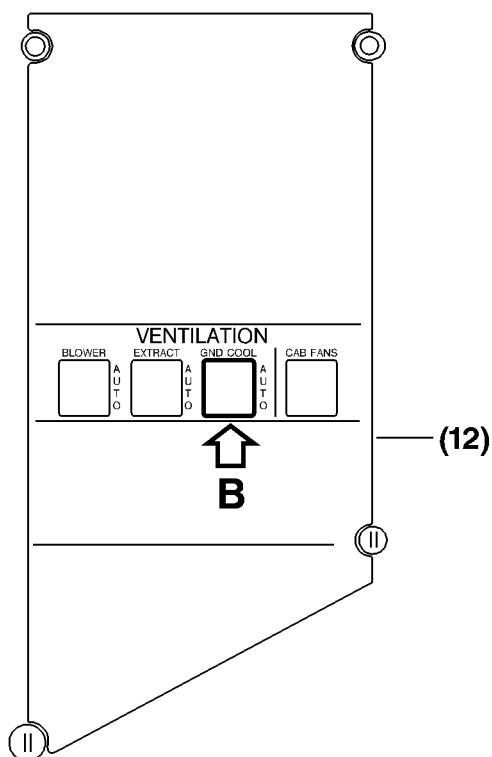
SERVICE BULLETIN



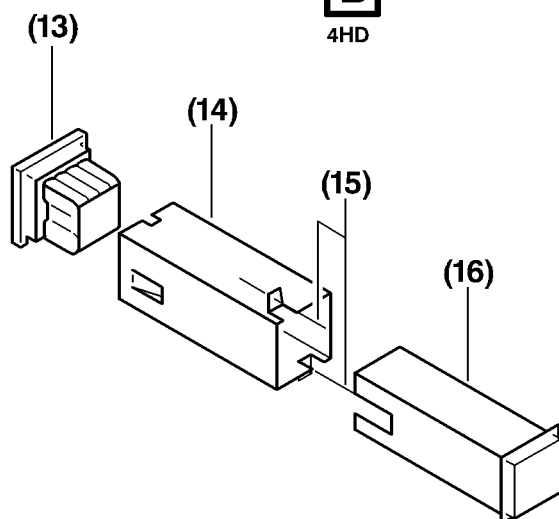
BEFORE



22VU



4HD



NSB5 211148 AN 00 d

Figure 9 Sheet 1
Modification of the Integrally Lighted Plate (9LF), on the Overhead Control Panel 22VU

DATE : Sep 27/04

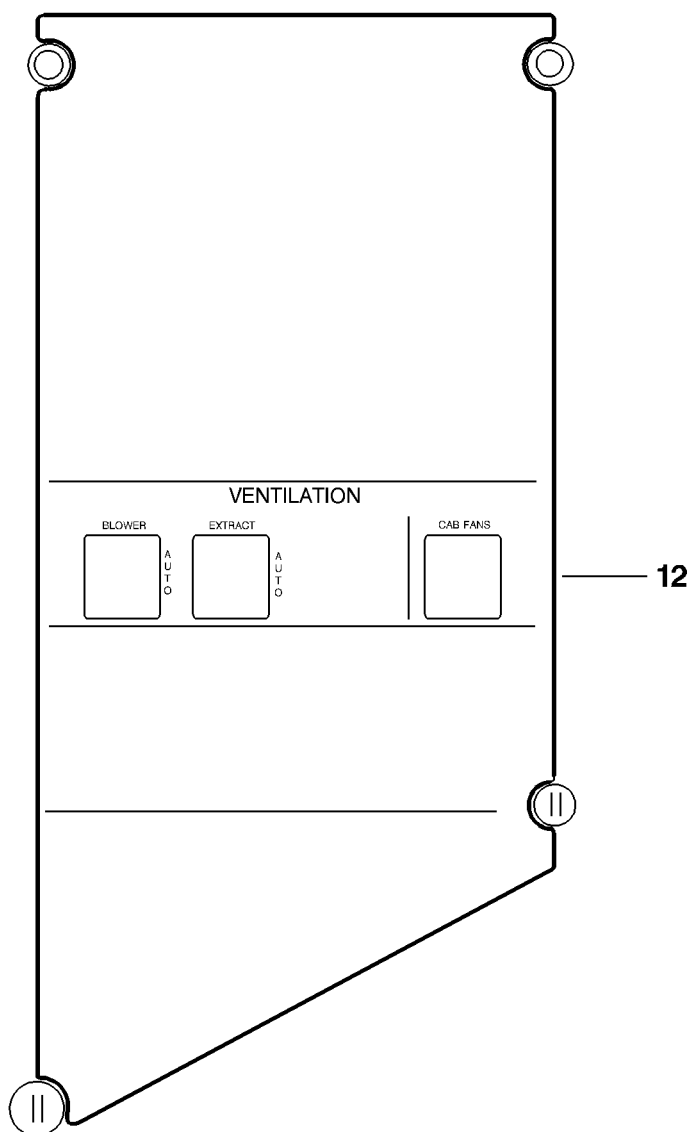
SERVICE BULLETIN No. : A320-21-1148

REVISION No. : 00 -

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AFTER



NSB5 211148 AO 00 d

Figure 9 Sheet 2
Modification of the Integrally Lighted Plate (9LF), on the Overhead Control Panel 22VU

DATE : Sep 27/04

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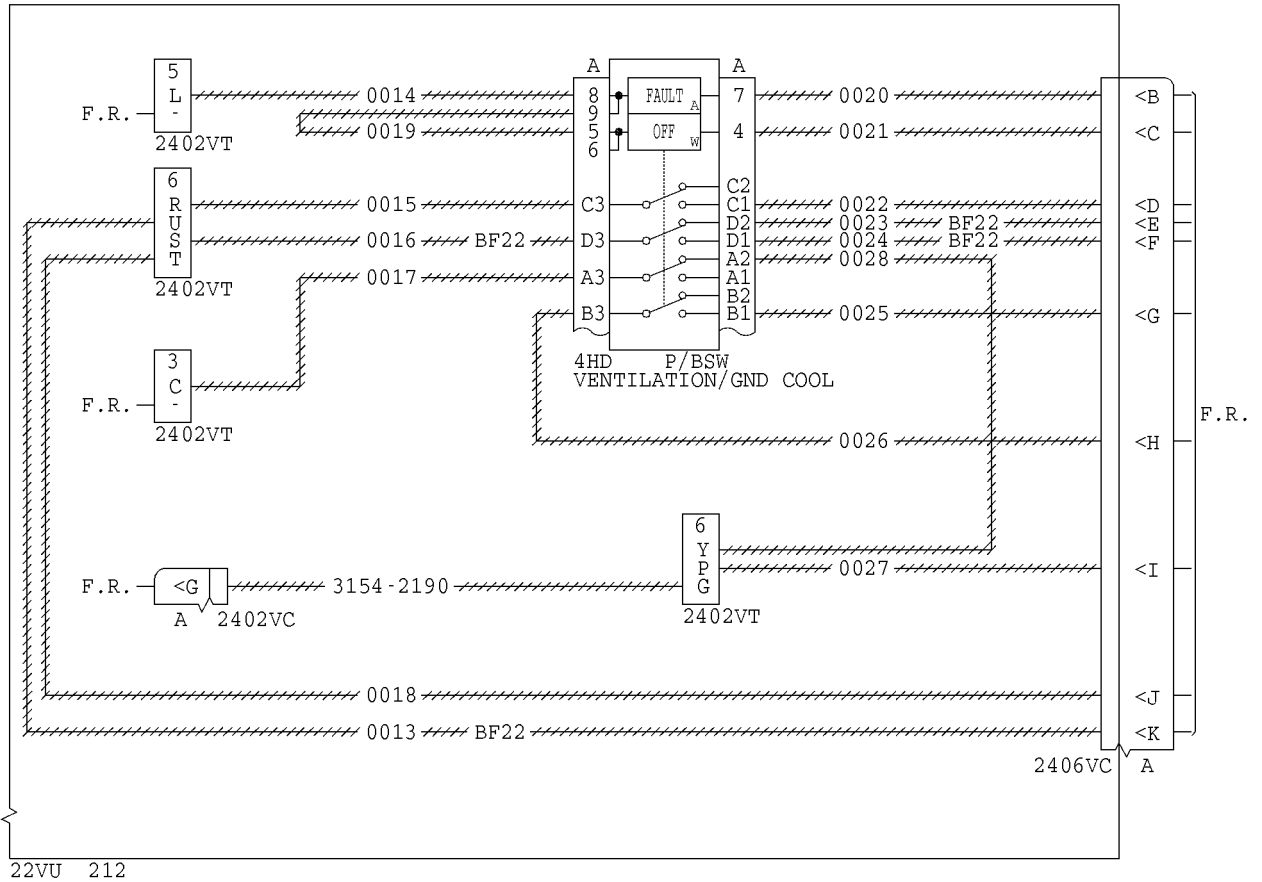
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SERVICE BULLETIN



F.R. FOR REFERENCE

DELETED WIRE

NOTE : UNLESS OTHERWISE SPECIFIED PREFIX ALL WIRE IDENTIFICATION WITH ATA2127
UNLESS OTHERWISE SPECIFIED ALL WIRES ARE BF24 GAUGE
UNLESS OTHERWISE SPECIFIED ALL ROUTES ARE 1M

NB5211148BAAAA

Figure 10 Sheet 1
Modification of the Wiring in the Overhead Control Panel 22VU (Config. 01)

DATE : Sep 27/04

SERVICE BULLETIN No. : A320-21-1148

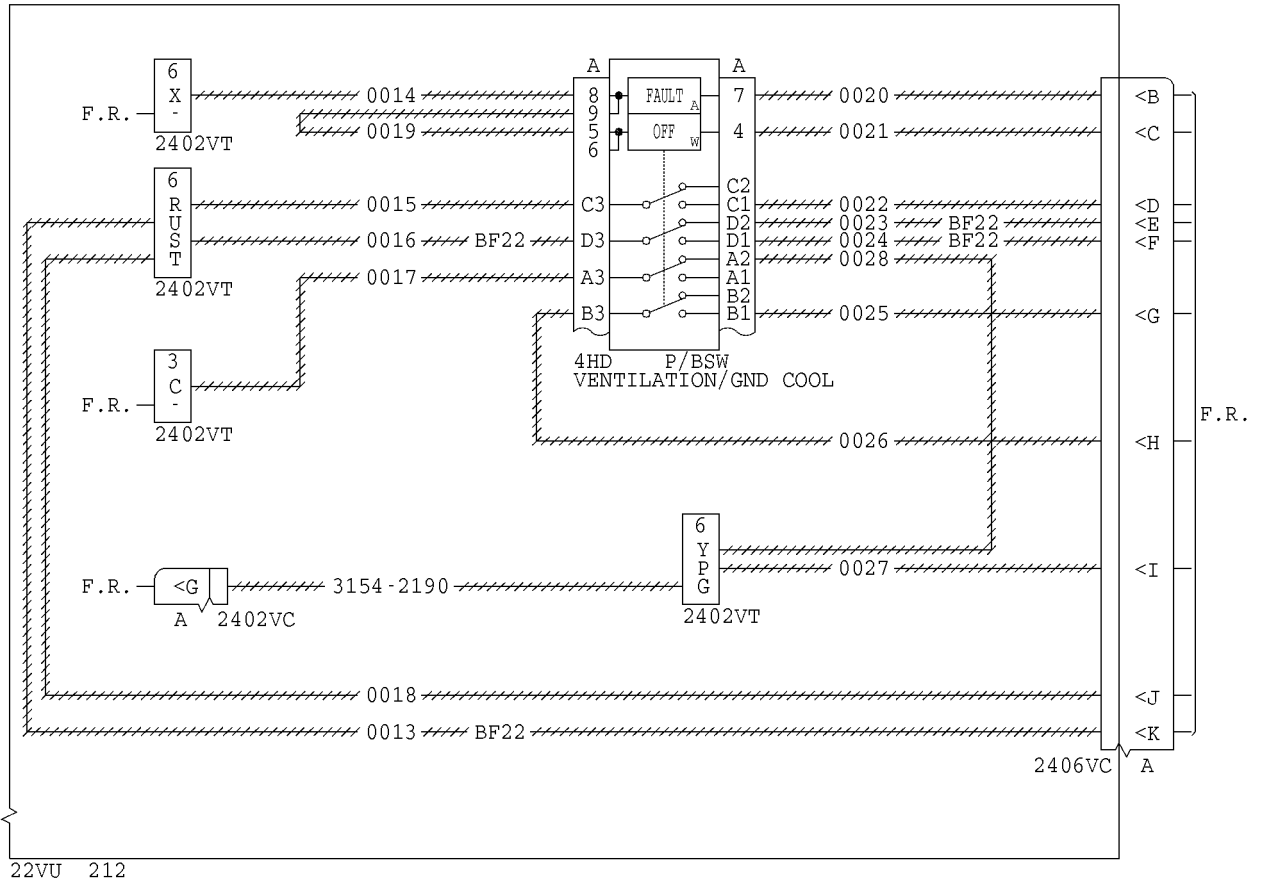
REVISION No. : 00 -

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SERVICE BULLETIN



F.R. FOR REFERENCE

DELETED WIRE

NOTE : UNLESS OTHERWISE SPECIFIED PREFIX ALL WIRE IDENTIFICATION WITH ATA2127
UNLESS OTHERWISE SPECIFIED ALL WIRES ARE BF24 GAUGE
UNLESS OTHERWISE SPECIFIED ALL ROUTES ARE 1M

NB5211148BBAAA

Figure 11 Sheet 1
Modification of the Wiring in the Overhead Control Panel 22VU (Config. 02)

DATE : Sep 27/04

SERVICE BULLETIN No. : A320-21-1148

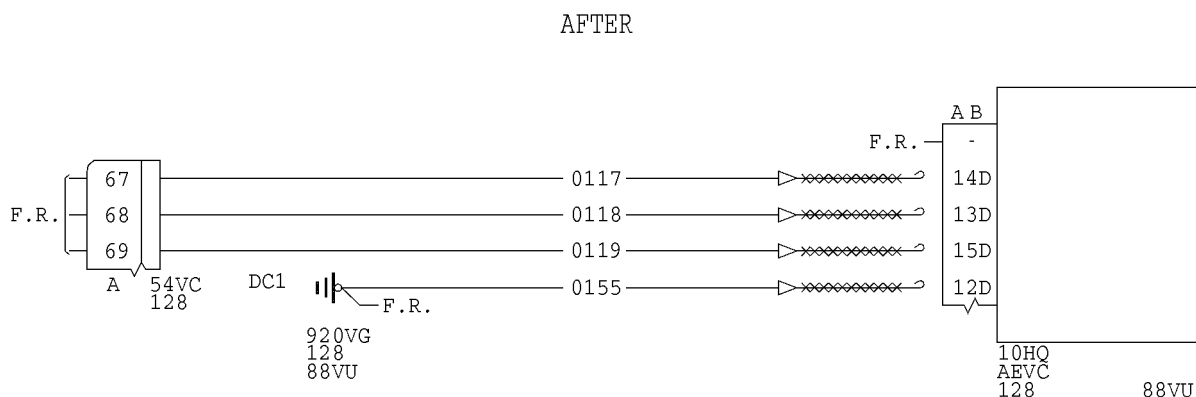
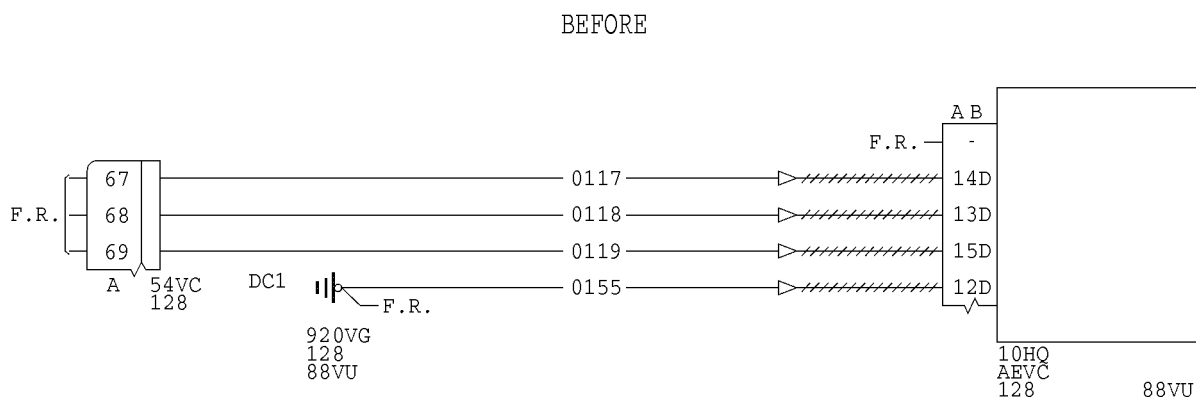
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SERVICE BULLETIN



— STOWED WIRE

△/△/△/△ OLD HOOK-UP

△××××× NEW HOOK-UP

F.R. FOR REFERENCE

NOTE : UNLESS OTHERWISE SPECIFIED PREFIX ALL WIRE IDENTIFICATION WITH ATA2127
UNLESS OTHERWISE SPECIFIED ALL WIRES ARE CF24 GAUGE
UNLESS OTHERWISE SPECIFIED ALL ROUTES ARE 1M

Figure 12 Sheet 1
Modification of the Wiring in the Shelf 88VU

DATE : Sep 27/04

SERVICE BULLETIN No. : A320-21-1148

REVISION No. : 00 –

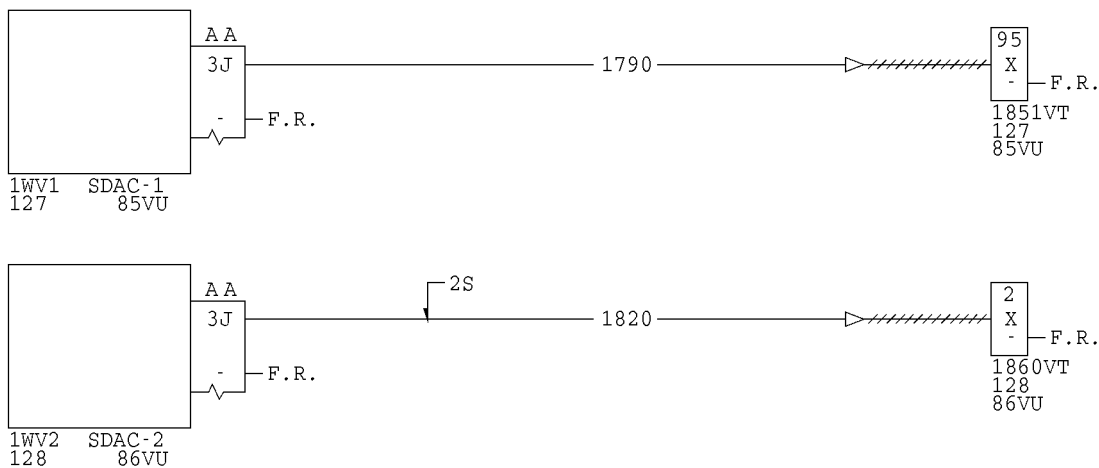
Page : 63



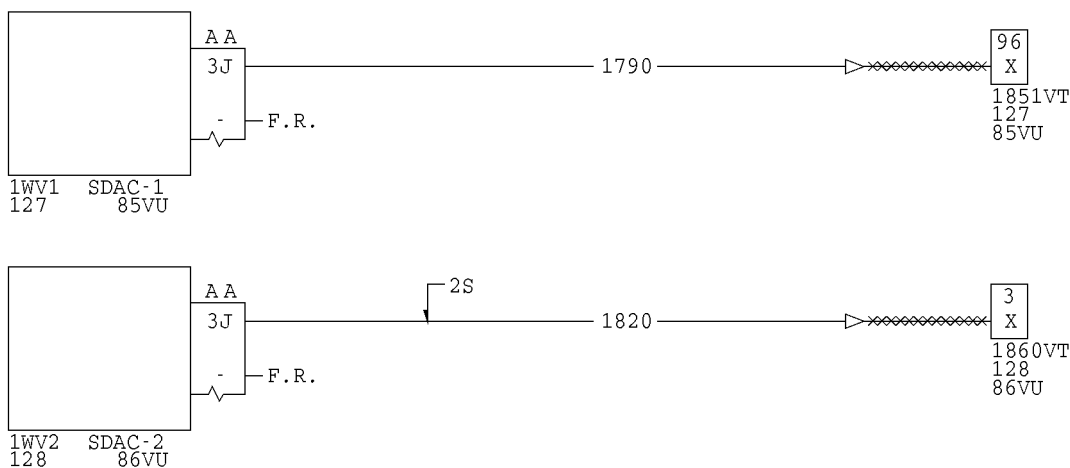
A318/A319/A320/A321

SERVICE BULLETIN

BEFORE



AFTER



OLD HOOK-UP

NEW HOOK-UP

F.R. FOR REFERENCE

NOTE : UNLESS OTHERWISE SPECIFIED PREFIX ALL WIRE IDENTIFICATION WITH ATA3154
UNLESS OTHERWISE SPECIFIED ALL WIRES ARE CF24 GAUGE
UNLESS OTHERWISE SPECIFIED ALL ROUTES ARE 1S

Figure 13 Sheet 1
Modification of the pin Programming of the SDACs

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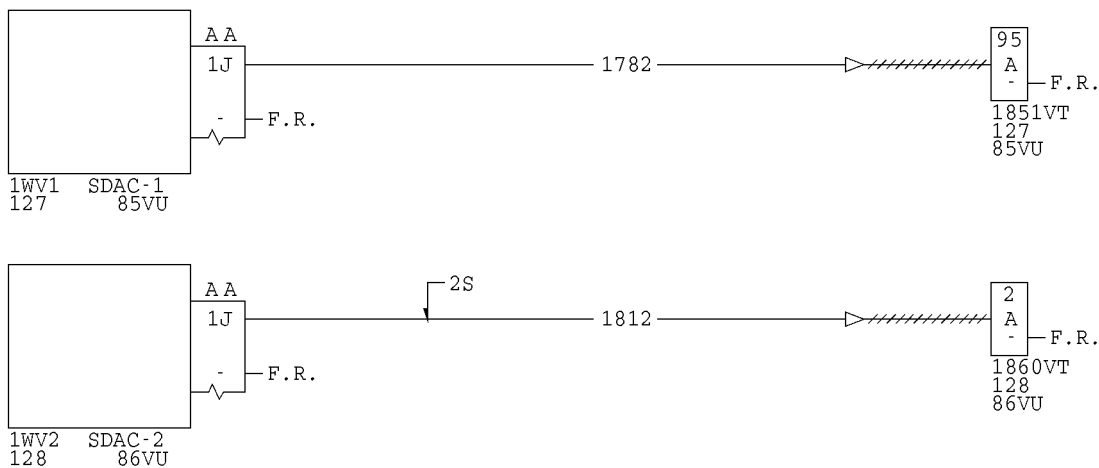
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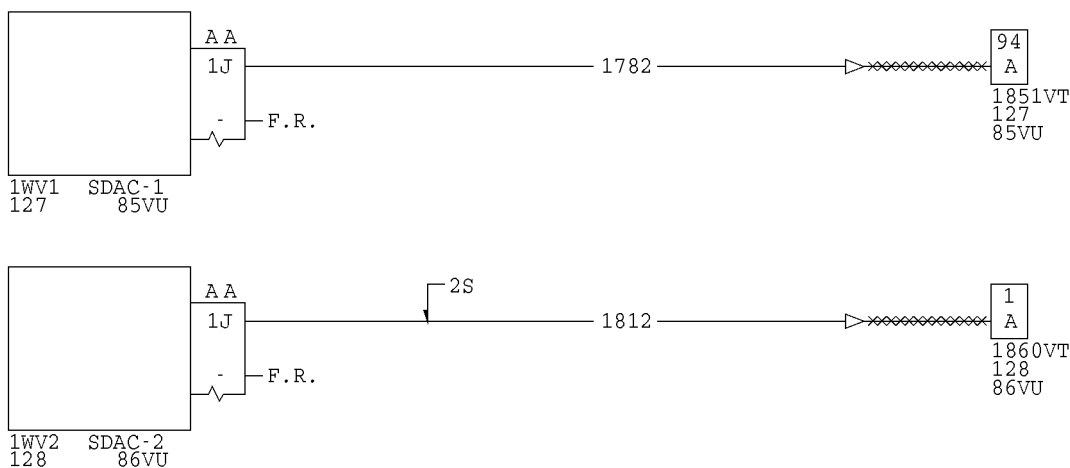
A318/A319/A320/A321

SERVICE BULLETIN

BEFORE



AFTER



△//// OLD HOOK-UP

△×××× NEW HOOK-UP

F.R. FOR REFERENCE

NOTE : UNLESS OTHERWISE SPECIFIED PREFIX ALL WIRE IDENTIFICATION WITH ATA3154
 UNLESS OTHERWISE SPECIFIED ALL WIRES ARE CF24 GAUGE
 UNLESS OTHERWISE SPECIFIED ALL ROUTES ARE 1S

NE5211148BEAAA

Figure 14 Sheet 1
 Modification of the Parity of the SDACs

DATE : Sep 27/04

SERVICE BULLETIN No. : A320-21-1148

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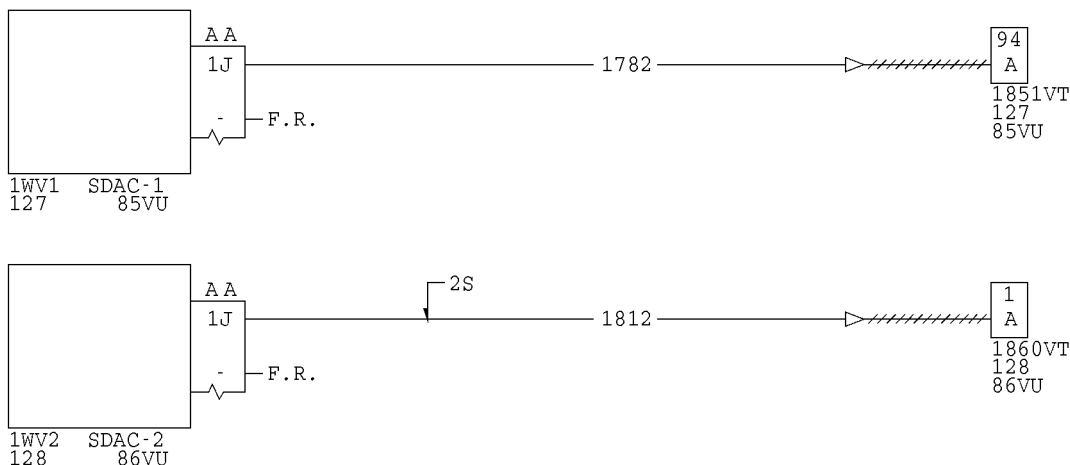
Page : 65



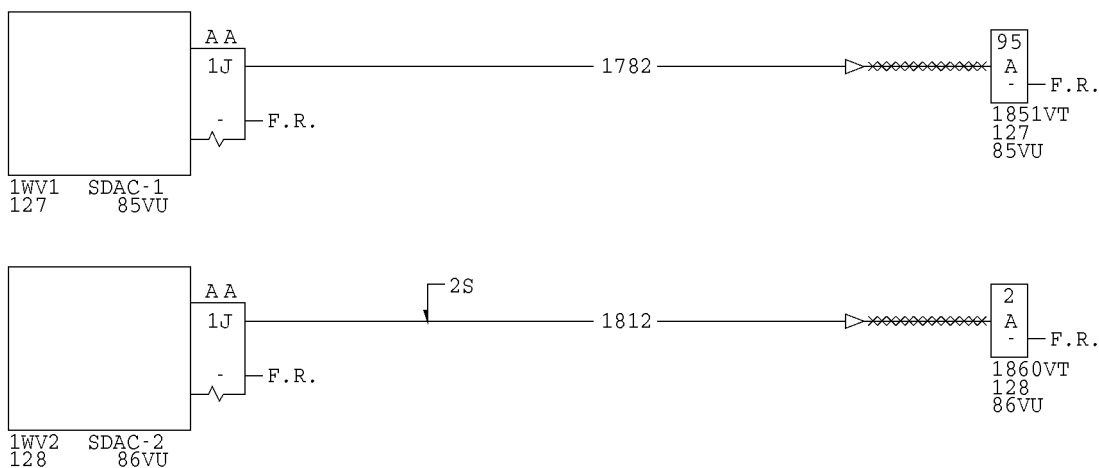
A318/A319/A320/A321

SERVICE BULLETIN

BEFORE



AFTER



OLD HOOK-UP

NEW HOOK-UP

F.R. FOR REFERENCE

NOTE : UNLESS OTHERWISE SPECIFIED PREFIX ALL WIRE IDENTIFICATION WITH ATA3154
UNLESS OTHERWISE SPECIFIED ALL WIRES ARE CF24 GAUGE
UNLESS OTHERWISE SPECIFIED ALL ROUTES ARE 1S

Figure 15 Sheet 1
Modification of the Parity of the SDACs

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Figure 16 Sheet 1 (Goto A3 section)
Hook-up Chart

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SERVICE BULLETIN

Figure 17 Sheet 1 (Goto A3 section)
Hook-up Chart

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SERVICE BULLETIN

Figure 18 Sheet 1 (Goto A3 section)
Hook-up Chart

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SERVICE BULLETIN

Figure 19 Sheet 1 (Goto A3 section)
Hook-up Chart

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SERVICE BULLETIN

Figure 20 Sheet 1 (Goto A3 section)
Hook-up Chart

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SERVICE BULLETIN

Figure 21 Sheet 1 (Goto A3 section)
Hook-up Chart

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SERVICE BULLETIN REPORTING SHEET

TITLE : AIR CONDITIONING - AVIONICS EQUIPMENT GROUND COOLING - DEACTIVATE THE
GROUND REFRIGERATION UNIT (GRU).

MODIFICATION No. : 33129P8139

Please complete the appropriate item (A or B):

A - SB WILL BE embodied YES/NO (if NO please comment)
If YES, aircraft concerned (as per SB effectivity by default) and planned dates
(month/year) of embodiment:

B - SB HAS BEEN embodied on aircraft:
.
.

Operator comments:
.
.

From Airline:
Name/Title:
Signature: Date:

If operational documentation is affected (see Paragraph 1.K of this SB): If
information is needed prior to next normal revision or prior to SB embodiment,
please indicate required service(s):

Either: Advance Data YES/NO
Or : Intermediate/Temporary revision. YES/NO

Important Information: This SB will only be incorporated in your maintenance and
operational documentation if this sheet is returned to Airbus and signed by a
duly authorised representative. With the next feasible revision, this will
result in

- updating of maintenance documentation to show pre and post SB data.
- updating of maintenance and operational documentation to show post SB data
after embodiment.

If this SB requires previous or simultaneous accomplishment of other SBs,
Airbus shall automatically include them in the manual revisions. Refer to SIL
00-037 for detailed information.

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Alternatively, SB lists via letters or fax are also accepted.

5 DATE : Sep 27/04

SERVICE BULLETIN No. : A320-21-1148

REVISION No. : 00 -

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SERVICE BULLETIN QUALITY PERCEPTION FORM

Use this form to tell us what is your perception of the quality of this Service Bulletin. The reported data that you provide us will be used to analyse areas of difficulties and to take corrective action to further improve the quality of our Service Bulletins.

We thank you for the time you have taken in completing this form.

(Please rate on a scale of 1 to 4, with 4 being the highest score)

- | | | | | |
|---|-------|---|---|---|
| - Quality rating of this SB | 4 | 3 | 2 | 1 |
| - Quality rating of the Accomplishment Instructions | 4 | 3 | 2 | 1 |
| - Quality rating of the Illustrations | 4 | 3 | 2 | 1 |
| - Is this SB easy to understand ? | Y / N | | | |

If you have had difficulties in the accomplishment of this SB please quote below the area(s) and give a short description of the issue.

Planning	Material	Instructions
X Effectivity	X Kit content	X Preparation
X Reason	X List of Materials	X Mod/Inspection
X Manpower	Operator Supplied	X Test
X References	X Re-identification	X Close-Up
X Publication	X Tooling	X Illustrations

Comments :

Operator :

Date:

Name/Title :

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1 Rond Point Maurice Bellonte
31707 BLAGNAC
Attn : SET1 Service Bulletins Management
FAX : (33) 5 61 93 42 51

Or via your Resident Customer Support Office.

5 DATE : Sep 27/04

SERVICE BULLETIN No. : A320-21-1148

REVISION No. : 00 -

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Line	E N D 1				L E A D						E N D 2				Instructions	
	Zone or Panel	Elec.Ident.	Term	Terminal P/N	Wire Ident.	Co	Rte	Gauge	Length		Zone or Panel	Elec.Ident.	Term	Terminal P/N		
									mm	Inch						
1	22VU	4HD-A	8		2127-0014			BF24			22VU	2402VT	5L		D	(1) FIG 10
2	22VU	4HD-A	C3		2127-0015			BF24			22VU	2402VT	6R		D	(1) FIG 10
3	22VU	4HD-A	D3		2127-0016			BF22			22VU	2402VT	6S		D	(1) FIG 10
4	22VU	4HD-A	A3		2127-0017			BF24			22VU	2402VT	3C		D	(1) FIG 10
5	22VU	4HD-A	9		2127-0019			BF24			22VU	4HD-A	5		D	(1) FIG 10
6	22VU	4HD-A	7		2127-0020			BF24			22VU	2406VC	<B		D	(1) FIG 10
7	22VU	4HD-A	4		2127-0021			BF24			22VU	2406VC	<C		D	(1) FIG 10
8	22VU	4HD-A	C1		2127-0022			BF24			22VU	2406VC	<D		D	(1) FIG 10
9	22VU	4HD-A	D2		2127-0023			BF22			22VU	2406VC	<E		D	(1) FIG 10
10	22VU	4HD-A	D1		2127-0024			BF22			22VU	2406VC	<F		D	(1) FIG 10
11	22VU	4HD-A	B1		2127-0025			BF24			22VU	2406VC	<G		D	(1) FIG 10
12	22VU	4HD-A	B3		2127-0026			BF24			22VU	2406VC	<H		D	(1) FIG 10
13	22VU	4HD-A	A2		2127-0028			BF24			22VU	2402VT	6Y		D	(1) FIG 10
14	22VU	2406VC	<K		2127-0013			BF22			22VU	2402VT	6U		D	(1) FIG 10
15	22VU	2406VC	<J		2127-0018			BF24			22VU	2402VT	6T		D	(1) FIG 10
16	22VU	2406VC	<I		2127-0027			BF24			22VU	2402VT	6P		D	(1) FIG 10
17	22VU	2402VT	6G		3154-2190			BF24			22VU	2402VC	<G		D	(17 FIG 10
18																
19																
20																
21																
22																
23																
24																
25																
D = DELETED WIRE																
(1) = FOR CONFIG.01																

Figure 16 Sheet 1
Hook-up Chart

Line	E N D 1				L E A D						E N D 2				Instructions	
	Zone or Panel	Elec.Ident.	Term	Terminal P/N	Wire Ident.	Co	Rte	Gauge	Length		Zone or Panel	Elec.Ident.	Term	Terminal P/N		
									mm	Inch						
1	22VU	4HD-A	8		2127-0014			BF24			22VU	2402VT	6X		D	(2) FIG 11
2	22VU	4HD-A	C3		2127-0015			BF24			22VU	2402VT	6R		D	(2) FIG 11
3	22VU	4HD-A	D3		2127-0016			BF22			22VU	2402VT	6S		D	(2) FIG 11
4	22VU	4HD-A	A3		2127-0017			BF24			22VU	2402VT	3C		D	(2) FIG 11
5	22VU	4HD-A	9		2127-0019			BF24			22VU	4HD-A	5		D	(2) FIG 11
6	22VU	4HD-A	7		2127-0020			BF24			22VU	2406VC	<B		D	(2) FIG 11
7	22VU	4HD-A	4		2127-0021			BF24			22VU	2406VC	<C		D	(2) FIG 11
8	22VU	4HD-A	C1		2127-0022			BF24			22VU	2406VC	<D		D	(2) FIG 11
9	22VU	4HD-A	D2		2127-0023			BF22			22VU	2406VC	<E		D	(2) FIG 11
10	22VU	4HD-A	D1		2127-0024			BF22			22VU	2406VC	<F		D	(2) FIG 11
11	22VU	4HD-A	B1		2127-0025			BF24			22VU	2406VC	<G		D	(2) FIG 11
12	22VU	4HD-A	B3		2127-0026			BF24			22VU	2406VC	<H		D	(2) FIG 11
13	22VU	4HD-A	A2		2127-0028			BF24			22VU	2402VT	6Y		D	(2) FIG 11
14	22VU	2406VC	<K		2127-0013			BF22			22VU	2402VT	6U		D	(2) FIG 11
15	22VU	2406VC	<J		2127-0018			BF24			22VU	2402VT	6T		D	(2) FIG 11
16	22VU	2406VC	<I		2127-0027			BF24			22VU	2402VT	6P		D	(2) FIG 11
17	22VU	2402VT	6G		3154-2190			BF24			22VU	2402VC	<G		D	(2) FIG 11
18																
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22																
23																
24																
25																
D = DELETED WIRE																
(2) = FOR CONFIG.02																

Figure 17 Sheet 1
Hook-up Chart

Line	E N D 1				L E A D						E N D 2				Instructions	
	Zone or Panel	Elec.Ident.	Term	Terminal P/N	Wire Ident.	Co	Rte	Gauge	Length		Zone or Panel	Elec.Ident.	Term	Terminal P/N		
									mm	Inch						
1	88VU	10HQ-AB	12D	NSA936604CA0931	2127-0155		1M	CF24			88VU	920VG	GND		M1O	FIG 12
2	88VU	10HQ-AB	WS		2127-0155		1M	CF24			88VU	920VG	GND		M1N	FIG 12
3	88VU	10HQ-AB	14D		2127-0117		1M	CF24			128	54VC	67		M1O	FIG 12
4	88VU	10HQ-AB	WS		2127-0117		1M	CF24			128	54VC	67		M1N	FIG 12
5	88VU	10HQ-AB	13D		2127-0118		1M	CF24			128	54VC	68		M1O	FIG 12
6	88VU	10HQ-AB	WS		2127-0118		1M	CF24			128	54VC	68		M1N	FIG 12
7	88VU	10HQ-AB	15D		2127-0119		1M	CF24			128	54VC	69		M1O	FIG 12
8	88VU	10HQ-AB	WS		2127-0119		1M	CF24			128	54VC	69		M1N	FIG 12
9																
10																
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23																
24																
25																
M1O = OLD HOOK-UP ON END 1 WS = WIRE STOWED																
M1N = NEW HOOK-UP ON END 1																
GND = GROUND																

Figure 18 Sheet 1
Hook-up Chart

Line	E N D 1				L E A D						E N D 2				Instructions	
	Zone or Panel	Elec.Ident.	Term	Terminal P/N	Wire Ident.	Co	Rte	Gauge	Length		Zone or Panel	Elec.Ident.	Term	Terminal P/N		
									mm	Inch						
1	85VU	1851VT	95X	EN3155-016M2222	3154-1790		1S	CF24			85VU	1WV1-AA	3J		M1O	FIG 13
2	85VU	1851VT	96X		3154-1790		1S	CF24			85VU	1WV1-AA	3J		M1N	FIG 13
3	86VU	1860VT	2X		3154-1820		2S	CF24			86VU	1WV2-AA	3J		M1O	FIG 13
4	86VU	1860VT	3X		3154-1820		2S	CF24			86VU	1WV2-AA	3J		M1N	FIG 13
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25																
M1O = OLD HOOK-UP ON END 1 M1N = NEW HOOK-UP ON END 1																

Figure 19 Sheet 1
Hook-up Chart



A318/A319/A320/A321

SERVICE BULLETIN

Line	E N D 1				L E A D						E N D 2				Instructions
	Zone or Panel	Elec.Ident.	Term	Terminal P/N	Wire Ident.	Co	Rte	Gauge	Length		Zone or Panel	Elec.Ident.	Term	Terminal P/N	
									mm	Inch					
1	85VU	1851VT	95A	EN3155-016M2222	3154-1782		1S	CF24			85VU	1WV1-AA	1J		M1O (3) FIG 14
2	85VU	1851VT	94A		3154-1782		1S	CF24			85VU	1WV1-AA	1J		M1N FIG 14
3	86VU	1860VT	2A		3154-1812		2S	CF24			86VU	1WV2-AA	1J		M1O (3) FIG 14
4	86VU	1860VT	1A		3154-1812		2S	CF24			86VU	1WV2-AA	1J		M1N FIG 14
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25															
M1O = OLD HOOK-UP ON END 1 (3) = REFER TO PARAGRAPH PREPARATION															
M1N = NEW HOOK-UP ON END 1 FOR THE PARITY															

Figure 20 Sheet 1
Hook-up Chart

Line	E N D 1				L E A D						E N D 2				Instructions
	Zone or Panel	Elec.Ident.	Term	Terminal P/N	Wire Ident.	Co	Rte	Gauge	Length		Zone or Panel	Elec.Ident.	Term	Terminal P/N	
									mm	Inch					
1	85VU	1851VT	94A	EN3155-016M2222	3154-1782		1S	CF24			85VU	1WV1-AA	1J		M1O (3) FIG 15
2	85VU	1851VT	95A		3154-1782		1S	CF24			85VU	1WV1-AA	1J		M1N FIG 15
3	86VU	1860VT	1A		3154-1812		2S	CF24			86VU	1WV2-AA	1J		M1O (3) FIG 15
4	86VU	1860VT	2A		3154-1812		2S	CF24			86VU	1WV2-AA	1J		M1N FIG 15
5															
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25															
M1O = OLD HOOK-UP ON END 1 (3) = REFER TO PARAGRAPH PREPARATION															
M1N = NEW HOOK-UP ON END 1 FOR THE PARITY															

Figure 21 Sheet 1
Hook-up Chart