AIRBUS INDUSTRIE
PRODUCT SUPPORT DIRECTORATE
1 Rond Point Maurice BELLONTE
31707 BLAGNAC CEDEX FRANCE
Tel : (33) 61-93-33-33
Telex : AIRBU 530526 F

A320

SERVICE BULLETIN SUMMARY

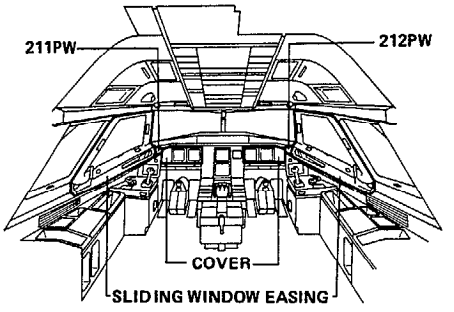
This summary is for information only and is not approved for modification of the aircraft.

MODIFICATION No. 23554P3127 ATA SYSTEM : 21

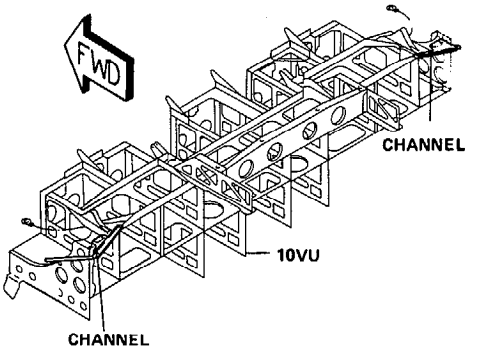
TITLE : AIR CONDITIONING - COCKPIT - INTRODUCE PROTECTION AGAINST WATER CONDENSATION.

REASON/DESCRIPTION/OPERATIONAL CONSEQUENCES

Some operators have reported damage and nuisance due to condensed water ; several locations have been identified mainly on the front window frame, on the sliding window casings on the cockpit, on the ceiling visible structures between frame 14 and frame 23 and at the lower and upper L and R windshield front sections.

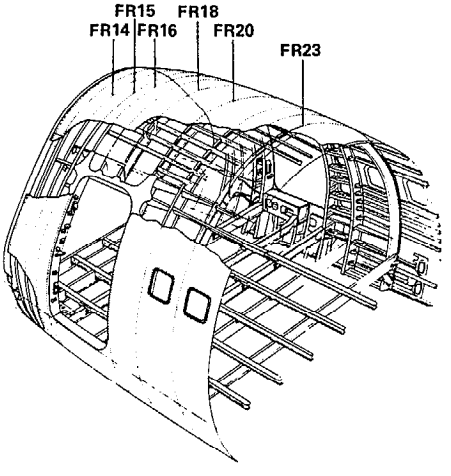


This Service Bulletin proposes to improve the protection against the condensation phenomenon in the cockpit.



The modification consists in :

- Adding a trough under the lower post of the front window frame at 301VU and 500VU.
- Adding a seal on the lower part of the sliding window easings.
- Insulating the ceiling visible fittings between frame 14 and frame 23.
- Blanking off the holes for the passage of frames 8 and 11 at the upper and lower windshield front sections.



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SERVICE BULLETIN SUMMARY

EFFECTIVITY

Kit No. 211072A01

This Service Bulletin is applicable to the following operator :

OYC.

NOTE : This Service Bulletin has been issued in response to the request of OYC. This modification as proposed in this Service Bulletin is possible on other aircraft.
Should other operators wish to embody this modification, please contact :

AIRBUS INDUSTRIE
PRODUCT SUPPORT DIRECTORATE
AI/SX DEPARTMENT
1 ROND POINT MAURICE BELLONTE
31707 BLAGNAC CEDEX
FRANCE

through RFC/RMO procedure.

SERVICE BULLETIN/MODIFICATION TO BE ACCOMPLISHED PREVIOUSLY OR SIMULTANEOUSLY

Service Bulletin No. A320-25-1092 (Mod. No. 23255P2936).

REFERENCES/REPERCUSSIONS

TFU	None	LIFE LIMIT	None
OEB	None	LINE MAINTENANCE AFFECTED	
AOT	None	NO	
SIL	None	OTHERS	None

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SERVICE BULLETIN SUMMARY

NATURE OF THE MODIFICATION

AIRCRAFT	YES
EQUIPMENT	NO
HARD	NO
SOFT	NO
OBRM	NO

COMPLIANCE

Optional.

MANPOWER

Manhours	75.0
Elapsed Time (Hours)	40.0

MATERIAL INFORMATION

Aircraft Data

Qty per A/C :

2 panel assies, 2 cover assies, 10 insulation blankets, 2 trough assies.

Equipment Data

None.

APPENDICES

None.

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A320

SERVICE BULLETIN

MODIFICATION No. 23554P3127 ATA SYSTEM : 21

TITLE : AIR CONDITIONING - COCKPIT - INTRODUCE PROTECTION AGAINST WATER CONDENSATION.

1. PLANNING INFORMATION

A. EFFECTIVITY

(1) Aircraft models : 320-231.

(2) Aircraft

Customer and Fleet No.	MSN	Kit No. 211072	Qty of Kits
OYC001-006	163,164,168,169,179,193	A01	06

This modification is applicable by Service Bulletin only on aircraft listed above.

NOTE : Accomplishment of this Service Bulletin requires the previous or simultaneous accomplishment of Service Bulletin No. A320-25-1092 (Mod. No. 23255P2936).

(3) Spares

None.

B. REASON

(1) History

Some operators have reported damage and nuisance due to condensed water ; several locations have been identified mainly on the front window frame, on the sliding window casings on the cockpit, on the ceiling visible structures between frame 14 and frame 23 and at the lower and upper L and R windshield front sections.

(2) Objective/Action

This Service Bulletin proposes to improve the protection against the condensation phenomenon in the cockpit.

The modification consists in :

- Adding a trough under the lower post of the front window frame at 301VU and 500VU.

- Adding a seal on the lower part of the sliding window easings.

- Insulating the ceiling visible fittings between frame 14 and frame 23.

- Blanking off the holes for the passage of frames 8 and 11 at the upper and lower windshield front sections.

(3) Advantages

To prevent water condensation in the cockpit.

(4) Accomplishment Timescale

In accordance with operator experience.

C. DESCRIPTION

Accomplishment of this Service Bulletin consists in carrying out the following jobs on the aircraft :

(1) In the cockpit, removal of VU panels for access.

(2) In the cockpit, removal of console for access.

(3) In the cockpit, removal of coat stowage for access.

(4) In the cockpit, removal of the instrument panel blanking plate.

(5) Installation of a trough under the lower post of the front window frame at 301VU and 500VU.

(6) Modification of drain line under L and R lower windshield front sections.

(7) Installation of the seal on the lower part of the sliding window easings.

(8) Installation of the insulation blankets on the visible structures.

(9) Installation of sealant on the structural holes at the lower and upper L and R windshield front sections.

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(10) In the cockpit, installation of the instrument panel blanking plate.

(11) In the cockpit, installation of the coat stowage.

(12) In the cockpit, installation of the console.

(13) In the cockpit, installation of the access VU panels.

D. APPROVAL

This Service Bulletin relates to the change(s) referenced on page 1 which has (have) been approved under the authority of DGAC Design Organisation Approval No. C 01.
The technical information contained in this Service Bulletin has been approved under the authority of DGAC Design Organisation Approval No. C 01.

E. MANPOWER

	Manhours
Gain access	25
Removal/installation of VU panels for access	4
Removal/installation of console for access	6
Removal/installation of coat stowage in zone 210 for access	4
Modification of the instrument panel blanking plate	4
Installation of a trough under the lower post of the front window frame at 301VU and 500VU	8
Modification of drain line under L and R lower windshield front sections	8
Installation of the seal on the lower part of the sliding window easings	4

	Manhours
Installation of the insulation blankets on the visible structures	6
Blanking of structural holes at the lower and upper L and R windshield front sections.	6
TOTAL MANHOURS	75
ELAPSED TIME (HOURS)	40

NOTE : This Service Bulletin assumes that the aircraft has been placed in a maintenance status. The manhours/elapsed time estimates do not include preparation for the modification, non-productive elapsed time, or administrative functions.

F. MATERIAL - COST AND AVAILABILITY

(1) Material

Operators with aircraft listed under paragraph 1.A.(2) above should submit a purchase order quoting this Service Bulletin to :

AIRBUS INDUSTRIE
SPARES SUPPORT CENTER
WEG BEIM JAEGER 150
D2000 HAMBURG 63
GERMANY

(2) Cost and availability

Subject to specific negotiation between operator(s) involved and Airbus Industrie in the frame of the RFC (or RFC/RMO) procedure.

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G. TOOLING - PRICE AND AVAILABILITY

None.

H. WEIGHT AND BALANCE

MEW : + 0.498 kg (+ 1.097 lb)

Effect on balance : + 3.272 kgm (+ 23.66 lb.ft)

I. REFERENCES

Aircraft Maintenance Manual : 06-41-52, 06-41-53, 12-34-24, 20-21-11, 23-51-00, 24-41-00, 25-11-51, 25-13-14, 23-23-42, 25-31-41, 31-50-00, 31-60-00, 34-48-00, 51-23-11, 56-12-11

Consumable Material List

Structural Repair Manual : 51-44-00

J. PUBLICATIONS AFFECTED

Illustrated Parts Catalog : 25-13-01-11, 25-13-01-37

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2. ACCOMPLISHMENT INSTRUCTIONS

A. GENERAL

(1) Preparation

(a) De-energize the aircraft electrical circuits (Ref. AMM 24-41-00, P. Block 201) and electrically ground the aircraft (Ref. AMM 12-34-24, P. Block 201).

(b) Install access platforms.

(c) Open access doors 831 and 841 (Ref. AMM 06-41-52).

(d) Remove Captain and First Officer seats as necessary (Ref. AMM 25-11-51, P. Block 401).

(e) Remove panels 211AW, 211BW, 211BZ, 211CW, 211DW, 211EW, 211FW, 211GW, 212AW, 212BW, 212BZ, 212CW, 212DW, 212EW, 212FW, 212GW, 212HW, 212JW and 212KW (Ref. AMM 06-41-53).

(f) Remove foot rest and sliding table assembly (Ref. AMM 25-13-14, P. Block 401).

(g) Remove ceiling panels in forward utility area (Ref. AMM 25-23-42, P. Block 401).

(h) Remove galley 6478MM and stowage 6576MM at location 2 (Ref. AMM 25-31-41, P. Block 401).

(2) Standard procedures

(a) To clean using Mat. No. 11-003 (Ref. CML).

(b) To torque standard threaded fasteners (Ref. AMM 20-21-11).

(c) To renew protective finish (Ref. AMM 51-23-11) using Mat. No. 16-001 and 16-002 (Ref. CML).

(d) To deburr the holes and refer to the drilling data (Ref. SRM 51-44-00).

(e) To shorten the rivets as necessary.

(f) To apply sealant Mat. No. 09-016 (Ref. CML).

(g) To apply paint Mat. No. 16-047 (Ref. CML).

(h) To blank off the gaps with Mat. No. 05-015 (Ref. CML).

B. MODIFICATION

(1) In the cockpit, removal of VU panels for access

Refer to Figure 1, Sheets 1 of 2 and 2

(a) Remove :

1 panel 17VU	(retain)
1 panel 16VU	(retain)
1 panel 15VU	(retain)
1 panel 101VU	(retain)
1 panel 130VU	(retain)
1 panel 131VU	(retain)
1 panel 700VU	(retain)
1 panel 301VU	(retain)
1 panel 500VU	(retain)
1 + 1 panel	Item 50 (retain)

with related components.

NOTE : Keep all the attachment hardware.

(2) In the cockpit, removal of the consoles

Refer to Figure 2

(a) Remove :

1 console 1VU	Item 52 (retain)
1 console 101VU	Item 53 (retain)
1 console 7VU	Item 54 (retain)
1 console 700VU	Item 55 (retain)
1 console 702VU	Item 56 (retain)
40 screws	Item 57 (retain)
40 washers	Item 58 (retain)

NOTE : Keep all the attachment hardware.

(3) In the cockpit, removal of the coat stowage for access

Refer to Figure 3

(a) Remove :

1 tube	Item 60 (retain)
1 front panel	Item 61 (retain)
1 upper panel	Item 62 (retain)
1 lower panel	Item 63 (retain)
1 side panel	Item 64 (retain)

NOTE : Keep all the attachment hardware.

(4) In the cockpit, removal of the instrument panel blanking plate

Refer to Figure 4

(a) Remove :

1 panel assy	Item (3) (discard)
1 panel assy	Item (4) (discard)
1 cover assy	Item (5) (discard)
1 cover assy	Item (6) (discard)

NOTE : Keep the related components and all the attachment hardware.

(5) Installation of a trough under the lower post of the front window frame at 301VU and 500VU

Refer to Figure 5, Sheets 1 of 3, 2 and 3

(a) On the structure 10VU, left and right sides, refer to the specified dimensions, make marks and drill the 3.2 mm (0.126 in.) holes to the intermediate diameter.

(b) Put the trough assies Items 14 and 15 in position on the structure 10VU. Align the hole that is on the trough and on the sides of the structure 10VU.

(c) Temporarily, attach the trough assies and drill the 4 mm (0.157 in.) holes to the final diameters.

(d) Remove the trough assies, clean and deburr.

(e) Apply sealant Mat. No. 09-016 (Ref. CML) and install :

1 trough assy	D9251277800000	Item 14
1 trough assy	D9251277800100	Item 15
2 + 2 screws	NAS1096-1-5	Item 17
1 + 1 screw	NAS1100-08-10	Item 19
2 + 2 washers	ASNA2397-6L	Item 20
1 + 1 washer	ASNA2397-8	Item 21
1 + 1 nut	MS21042-08	Item 23
2 + 2 nuts	MS21042-06	Item 24

(6) Modification of the drain line under the L and R lower windshield front sections

Refer to Figure 6, Sheets 1 of 2 and 2

(a) On FR3 and FR4, left and right sides, refer to the specified dimensions, put in position and drill the 3.2 mm (0.126 in.) dia. holes on the structure to match the holes on the bracket Item 16 (Details B and C).

(b) Apply sealant Mat. No. 09-016 (Ref. CML) and install :

2 + 2 brackets	ASNA2328-3-4	Item 16
6 + 6 rivets	ASNA2050DCJ3215	Item 27

(c) Cut out the insulation blankets to install the brackets Item 16 and apply the adhesive material KB6 on the cut sections.

(d) On the brackets Item 16, install :

2 + 2 screws	NAS1096-3-10	Item 18
4 + 4 washers	ASNA2397-10L	Item 22
2 + 2 clamps	NSA5515-7	Item 25

(e) On the left and right sides, put in position and attach along the routing installed before :

1 + 1 pipe	A2181060426800	Item 1

with :

Conduit	E0432A16	Item 28
Adhesive tape	ASNA35622420	Item 30

(f) Connect the pipe Item 1 to the trough assies Items 14 and 15 and attach with :

1 + 1 clamp	ASNA0033-022	Item 26

(g) On the pipes that are in the aircraft, on the left and right sides, Item (35) cut and install :

1 + 1 tee D0003002400100 Item 2

NOTE : If necessary, remove and retain the fasteners of the drain line.

(h) Identify the modified items :

D21210698212 Item (35) becomes A2181060420200 Item 31 and D2121069821100 Item 32.

(i) Connect the pipes Items 1, 31 and 32 to the tee union, Item 2 and attach with :

3 + 3 clamps ASNA0033-022 Item 26

(7) Installation of the seal on the lower part of the sliding window easings

Refer to Figure 7

(a) Remove on the left and right sides (Ref. AMM 56-12-11, P. Block 401) :

1 carter assy Item 33
1 carter assy Item 34

NOTE : Keep the related components and all the attachment hardware.

(b) Refer to the specified dimensions and install :

Silicone strip ABS5006-4D Item 29

(c) Paint the specified area with Mat. No. 16-047 (Ref. CML).

NOTE : Protect the silicone strips.

(d) Identify the modified items :

D2511105300200 Item (33) becomes D2511158400000 Item 33.

D2511105300300 Item (34) becomes D2511158400100 Item 34.

(e) Install the carter assies Items 33 and 34 with the related components and with all the hardware retained at removal.

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(8) Installation of the insulation blankets on the visible structures

Refer to Figure 8

(a) On the ceiling and trim panel pick-up bracket, at FR14, install :

Refer to Figure 8, Sheets 1 of 7 and 2

1 insulation blanket	D2581053400000	Item 7
1 insulation blanket	D2581053500000	Item 8

(b) At the upper passenger door box continuity fittings between FR14 and FR15, install :

Refer to Figure 8, Sheets 1 of 7 and 3

1 + 1 insulation blanket	D2581053600000	Item 9

(c) On the ceiling panel pick-up bracket, at FR16, install :

Refer to Figure 8, Sheets 1 of 7 and 4

1 insulation blanket	D2581053700000	Item 10
1 insulation blanket	D2581053800000	Item 11

(d) On the ceiling panel pick-up bracket, at FR18, install :

Refer to Figure 8, Sheets 1 of 7 and 5

1 + 1 insulation blanket	D2581053900000	Item 12

(e) On the ceiling panel pick-up bracket, at FR20, install :

Refer to Figure 8, Sheets 1 of 7 and 6

1 insulation blanket	D2581053900000	Item 12

(f) On the fitting at stringer 6 between FR23, FR24, R side, install :

Refer to Figure 8, Sheets 1 of 7 and 7

1 insulation blanket	D2581054000000	Item 13

NOTE : These insulation blankets will be attached to the primary insulation by self-adhesive patches.

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(9) Installation of sealant on the structural holes at the lower and upper L and R windshield front sections

Refer to Figure 9, Sheets 1 of 3, 2 and 3

At the frames 8 and 11 intersection holes, install sealant Mat. No. 09-016.

(10) In the cockpit, installation of the instrument panel blanking plate

Refer to Figure 4

(a) Install :

1 panel assy	D2511157000000	Item 3
1 panel assy	D2511157000100	Item 4
1 cover assy	D2511158100000	Item 5
1 cover assy	D2511158100100	Item 6

with all the hardware retained at removal and with all the related components.

(11) In the cockpit, installation of the coat room

Refer to Figure 3

(a) Install :

1 tube	Item 60
1 front panel	Item 61
1 upper panel	Item 62
1 lower panel	Item 63
1 side panel	Item 64

with all the hardware retained at removal and with all the related components.

NOTE : Items 60, 61, 62, 63 and 64 were retained at removal.

(12) In the cockpit, installation of the console

Refer to Figure 2

(a) Install :

1 console 1VU	Item 52
1 console 101VU	Item 53
1 console 7VU	Item 54
1 console 700VU	Item 55

1 console 702VU	Item 56
40 screws	Item 57
40 washers	Item 58

with all the hardware retained at removal.

NOTE : Items 52, 53, 54, 55, 56, 57 and 58 were retained at removal.

(13) In the cockpit, installation of the access VU panels

Refer to Figure 1

(a) Install :

1 panel 17VU	
1 panel 16VU	
1 panel 15VU	
1 panel 101VU	
1 panel 130VU	
1 panel 131VU	
1 panel 700VU	
1 panel 301VU	
1 panel 500VU	
1 + 1 panel	Item 50

with the related components and with all the hardware retained at removal.

NOTE : The above panels as well as panel Item 50 were retained at removal.

C. TESTS

(1) For 2VU, 6VU, 301VU, 500VU, do an operational test of the audio management (Ref. AMM 23-51-00, P. Block 501) (Do only the part B of this procedure for the capt station).

(2) For 301VU and 500VU, do an operational test of the GPWS ground self test function (Ref. AMM 34-48-00, P. Block 501).

(3) For 301VU and 500VU, do an operational test of the EIS' switching functions (Ref. AMM 31-60-00, P. Block 501, EFIS' part only) (Do not do parts 4C, 4D and 4E).

(4) For 130VU and 131VU, do an operational test of the FWC (Ref. AMM 31-50-00, P. Block 501) and the test below.

Test of the CHRONO function on the ND (Navigation Display) :

Perform an EIS Start Procedure (Ref. AMM 31-60-00, P. Block 201).

ACTION	RESULT
Push the CHRONO pushbutton switch.	On the ND display the chrono counter starts.
Push the CHRONO pushbutton switch.	On the ND display the chrono counter stops.
Push the CHRONO pushbutton switch.	On the ND display the chrono counter shows 0.

D. CLOSE-UP

(1) Make certain that the work areas are clean and clear of tools and miscellaneous items of equipment.

(2) Install galley 6478MM and stowage 6576MM at location 2 (Ref. AMM 25-31-41, P. Block 401).

(3) Install forward utility area ceiling panels (Ref. AMM 25-23-42, P. Block 401).

(4) Install foot rest and sliding table assembly (Ref. AMM 25-13-14, P. Block 401).

(5) Install panels 211AW, 211BW, 211BZ, 211CW, 211DW, 211EW, 211FW, 211GW, 212AW, 212BW, 212BZ, 212CW, 212DW, 212EW, 212FW, 212GW, 212HW, 212JW and 212KW (Ref. AMM 06-41-53).

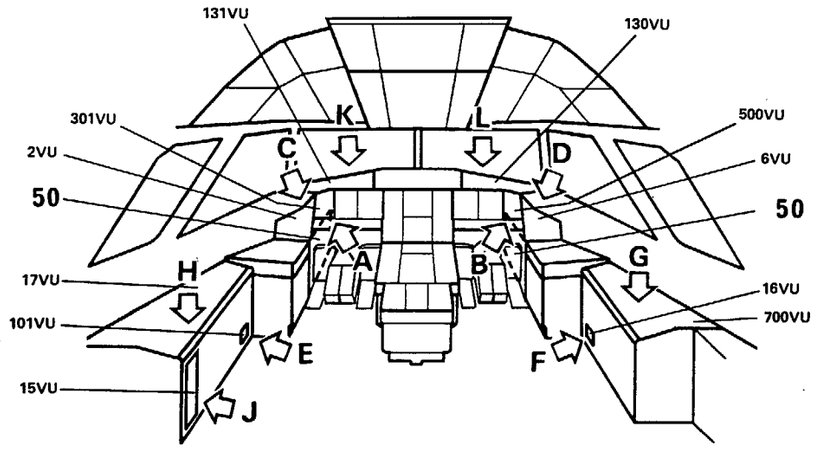
(6) Install Captain and First Officer seats if removed (Ref. AMM 25-11-51, P. Block 401).

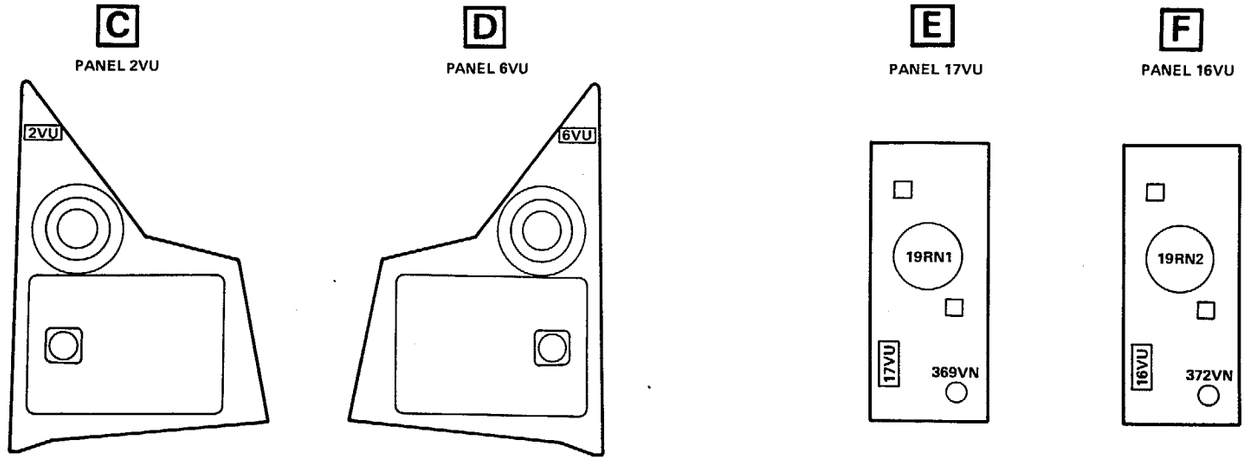
(7) Close access doors 831 and 841 (Ref. AMM 06-41-52).

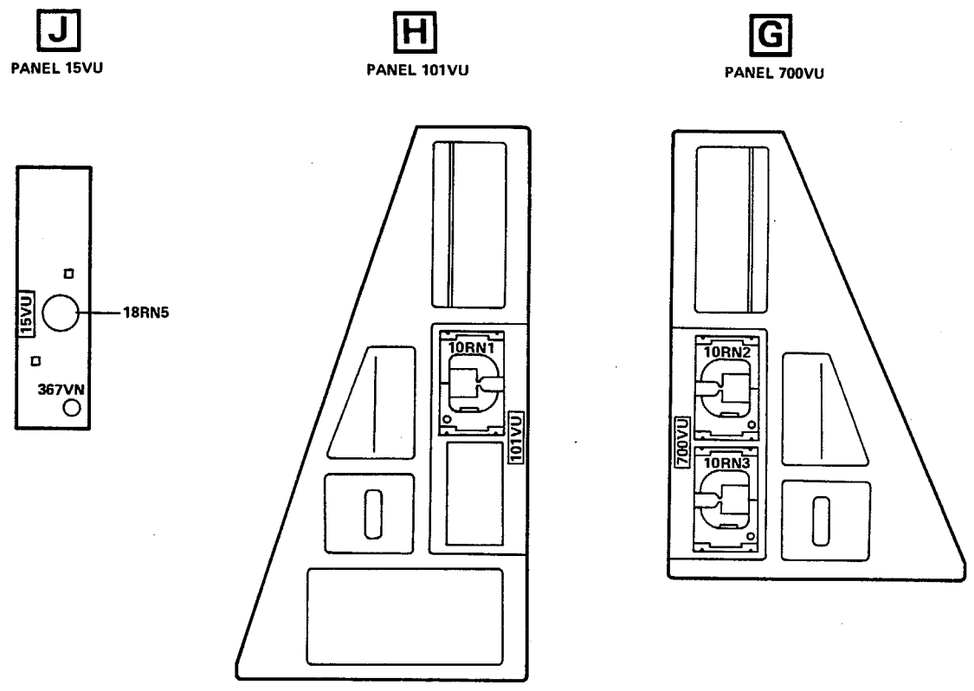
(8) Remove access platforms.

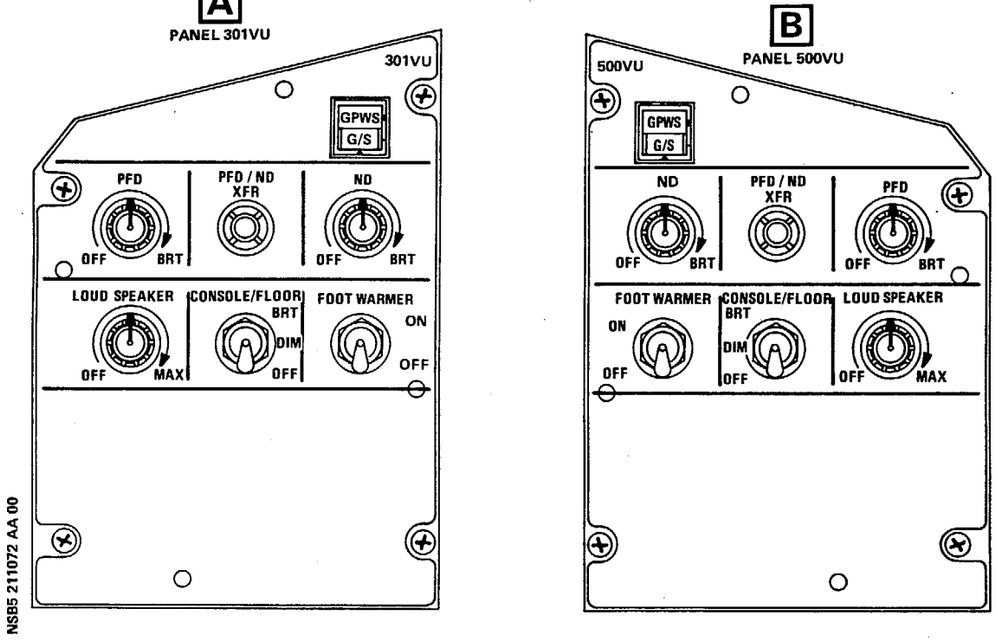
(9) Disconnect the aircraft electrical ground connection (Ref. AMM 12-34-24, P. Block 201).

(10) Write in the applicable aircraft record that you have done all the work in this Service Bulletin.









In cockpit, Removal/Installation of VU Panels for Access
Figure 1, Sheet 1 of 2

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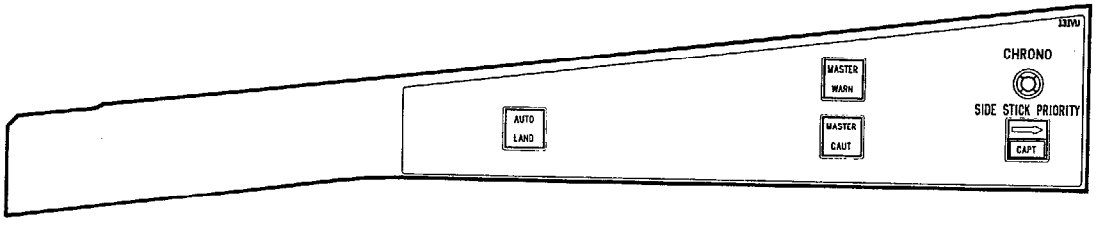
PANEL 131VU



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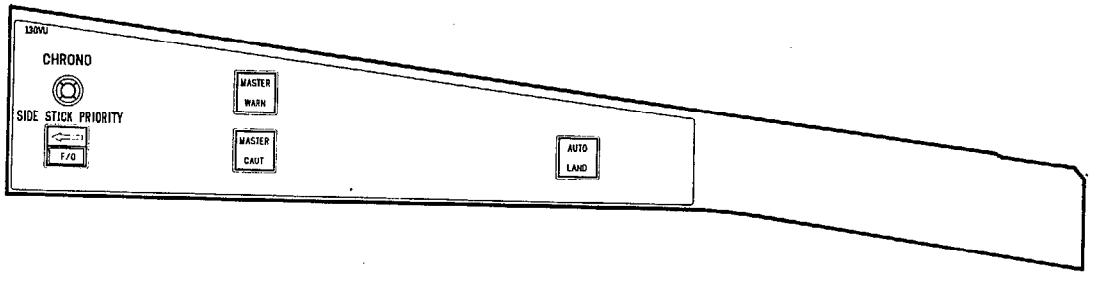
PANEL 130VU



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In Cockpit, Removal/Installation of VU Panels for Access
Figure 1, Sheet 2

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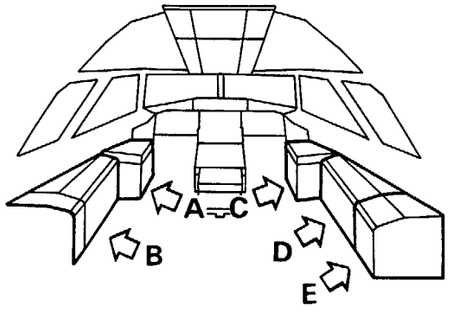

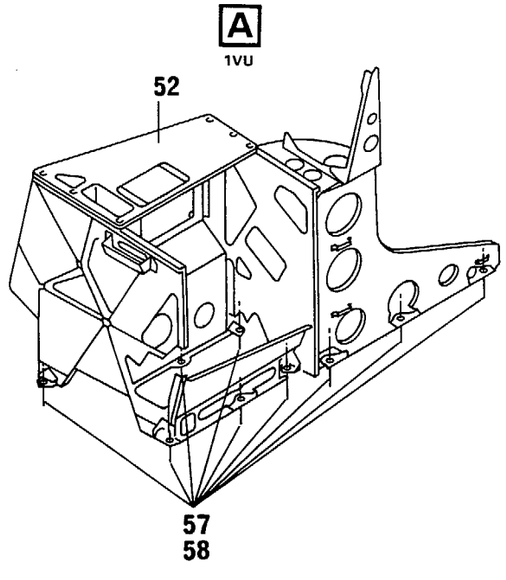

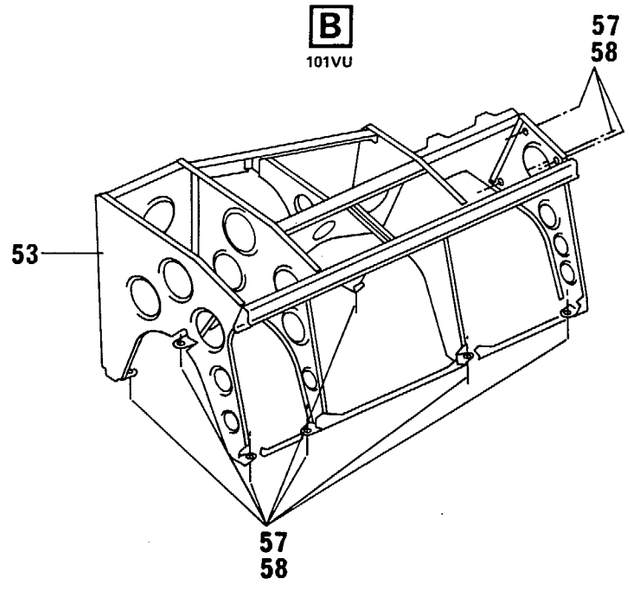

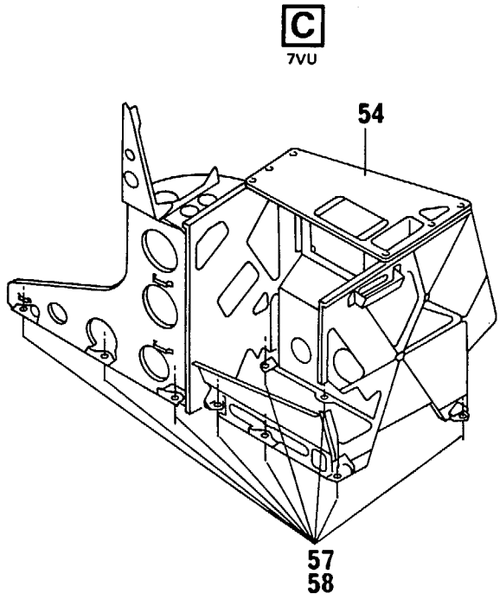

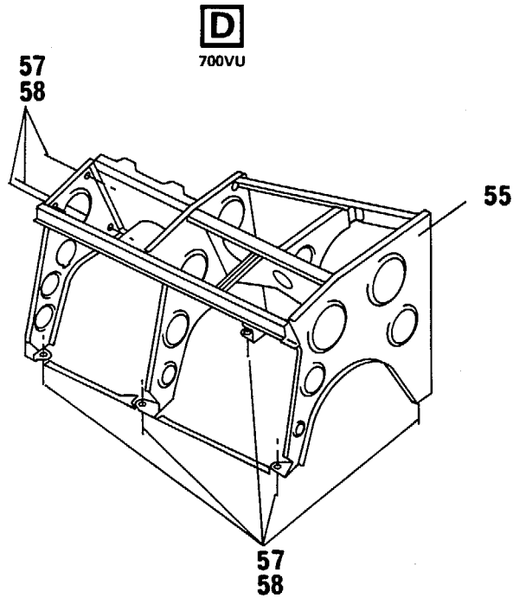

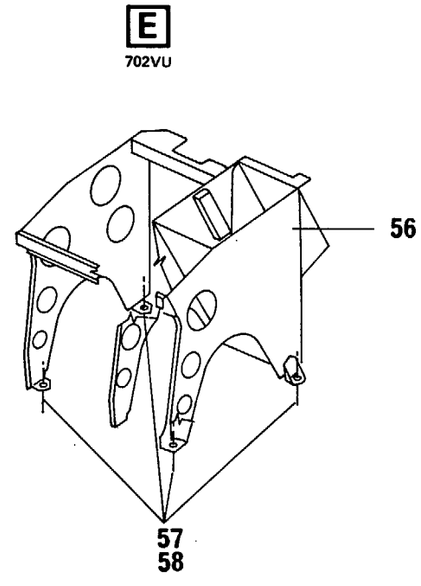

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In Cockpit, Removal/Installation of Console for Access
Figure 2

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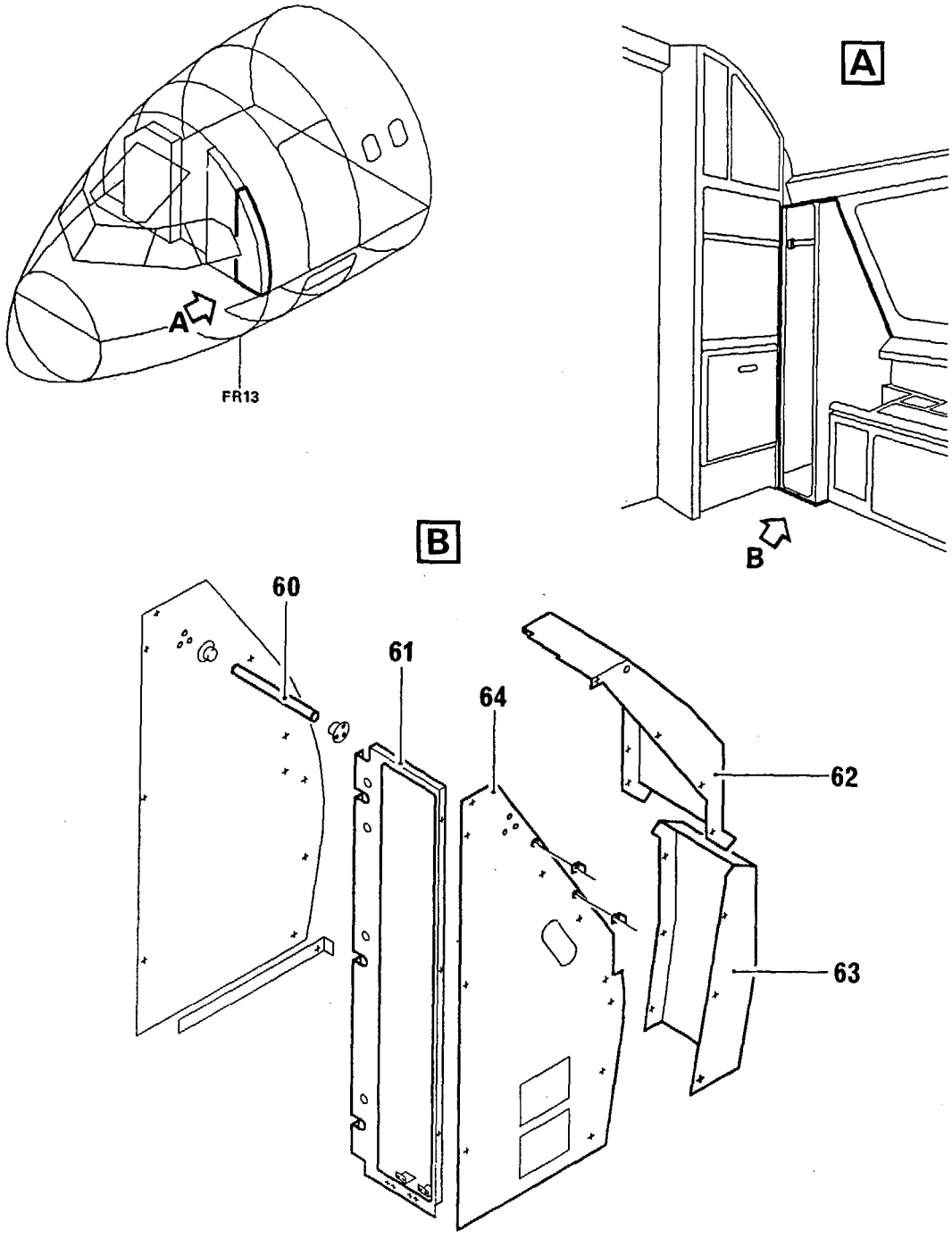

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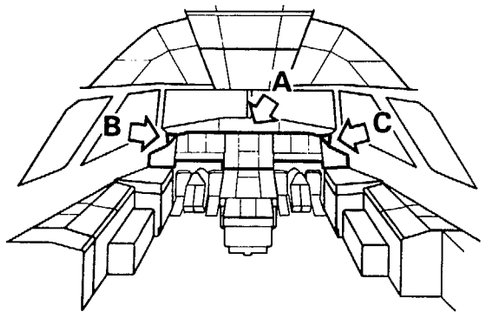
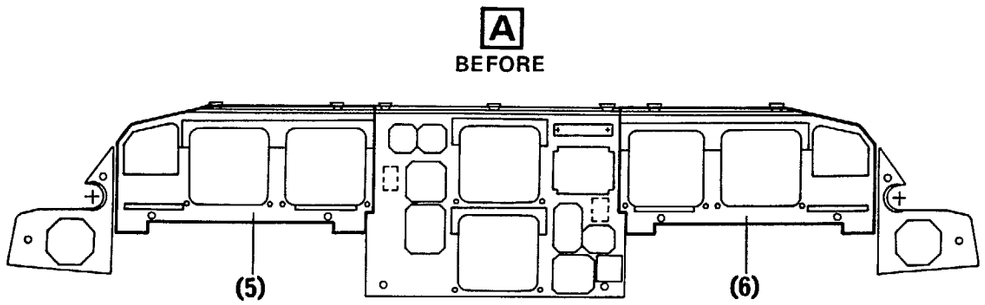
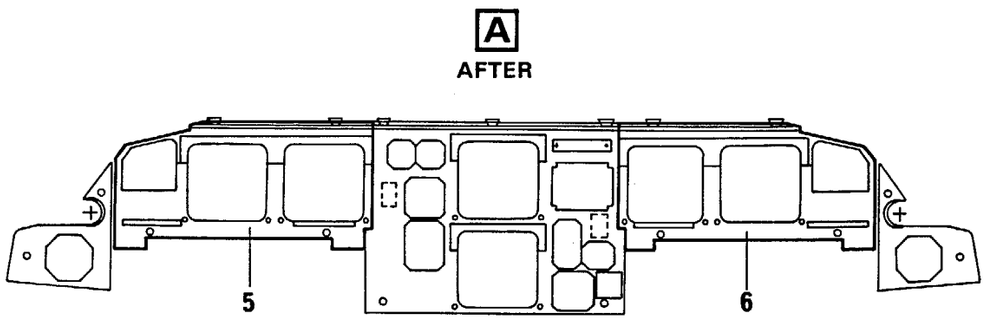
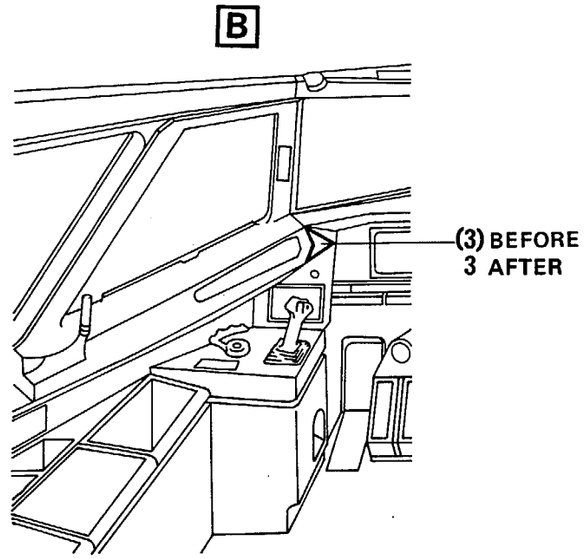
Removal/Installation of Coat Stowage in Zone 210 for Access
Figure 3

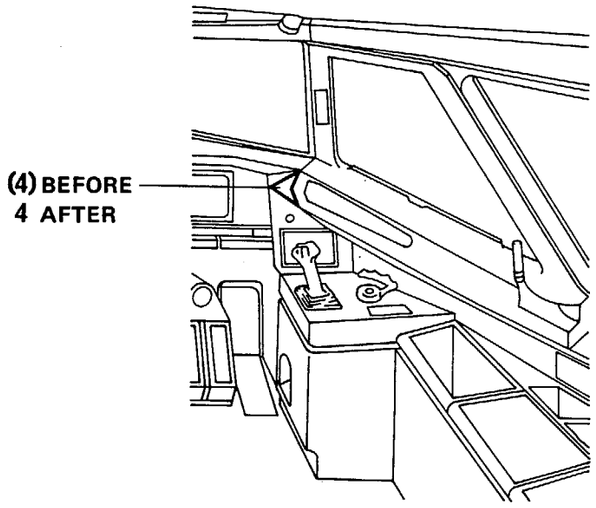






NSB5 211072 AI 00





Modification of the Instrument Panel Blanking Plate
Figure 4

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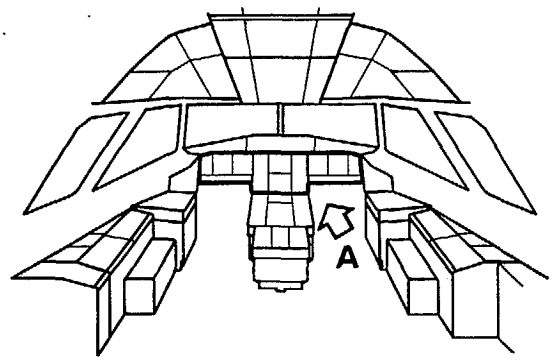
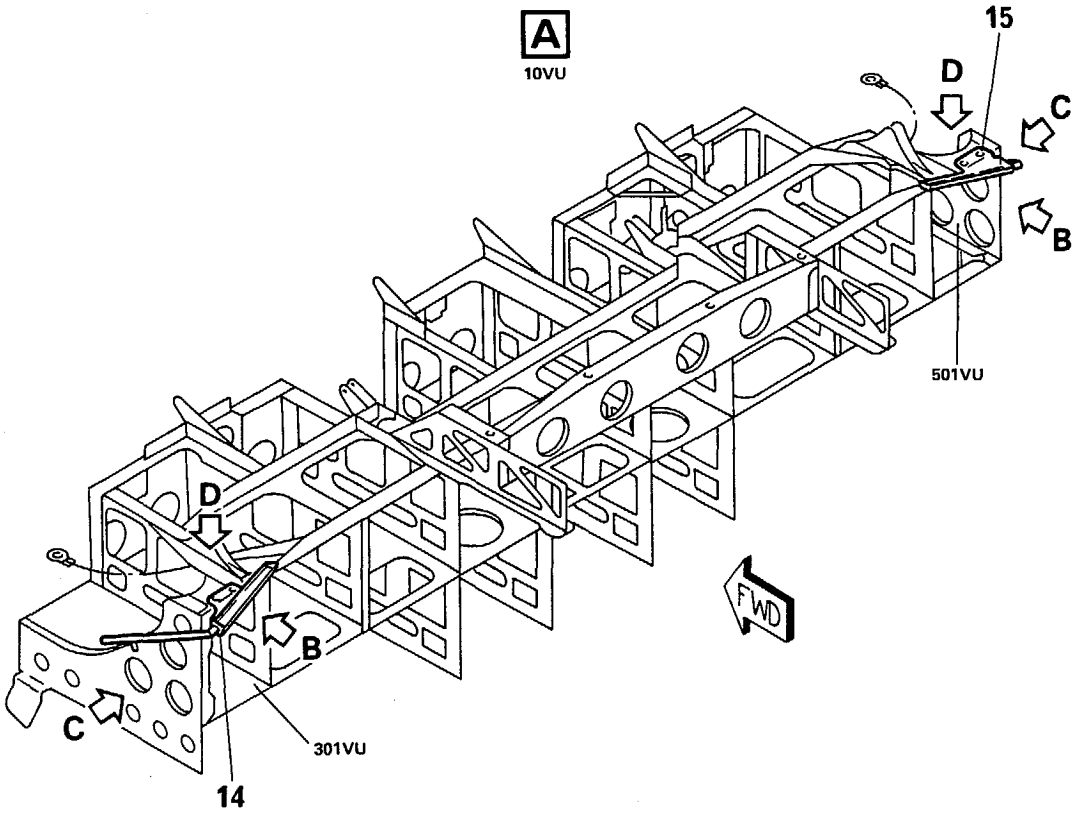
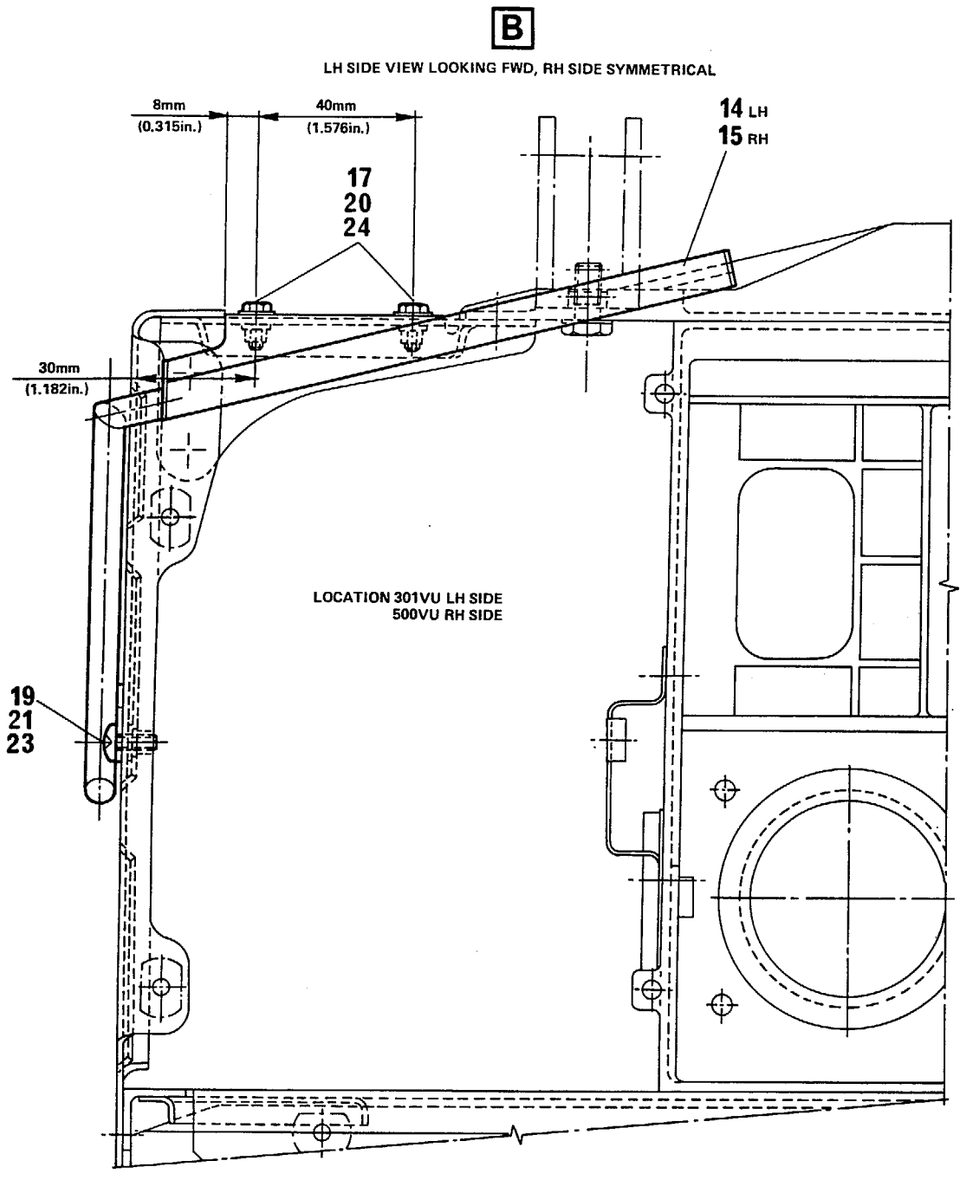


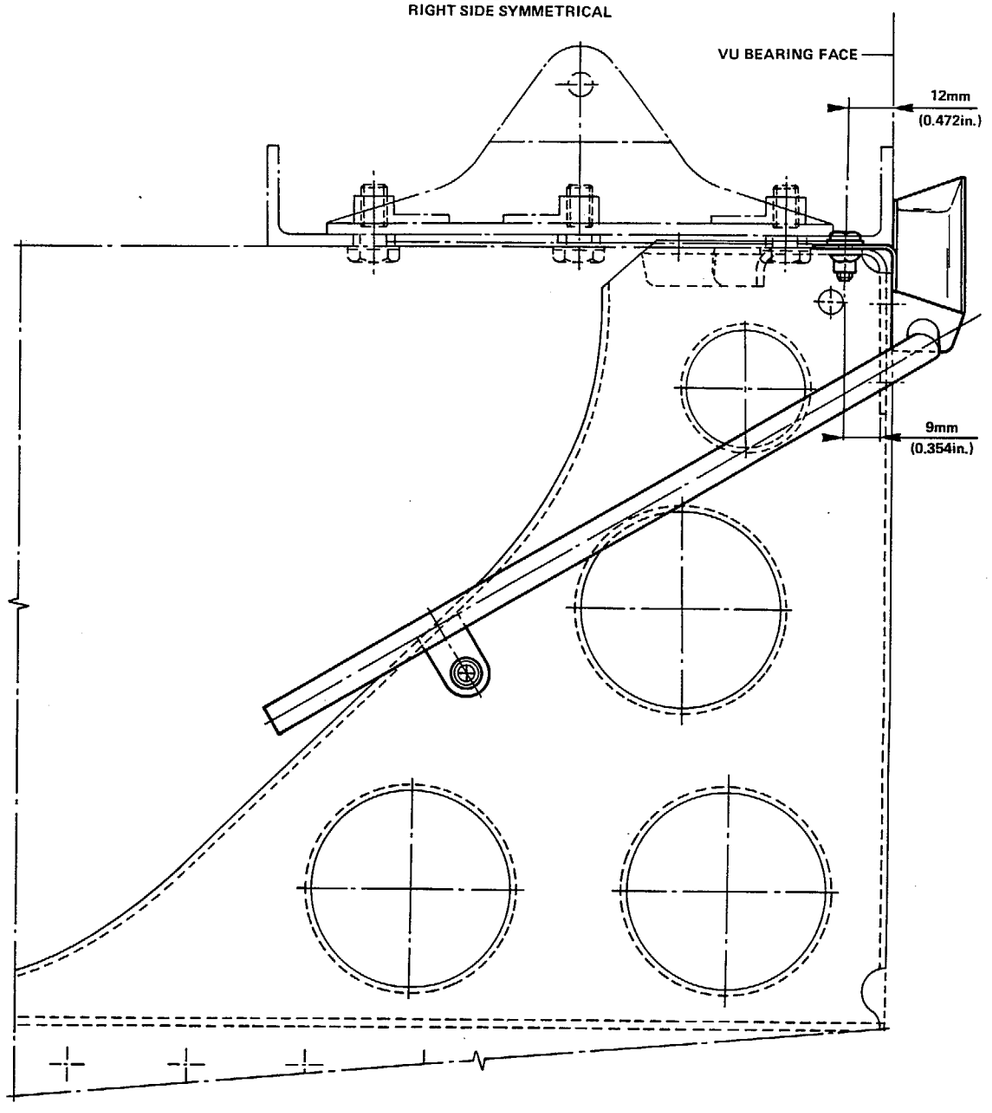


NSB5 211072 AD 00

Installation of a Trough under the Lower Post of the Front Window Frame at 301VU and 500VU
Figure 5, Sheet 1 of 3

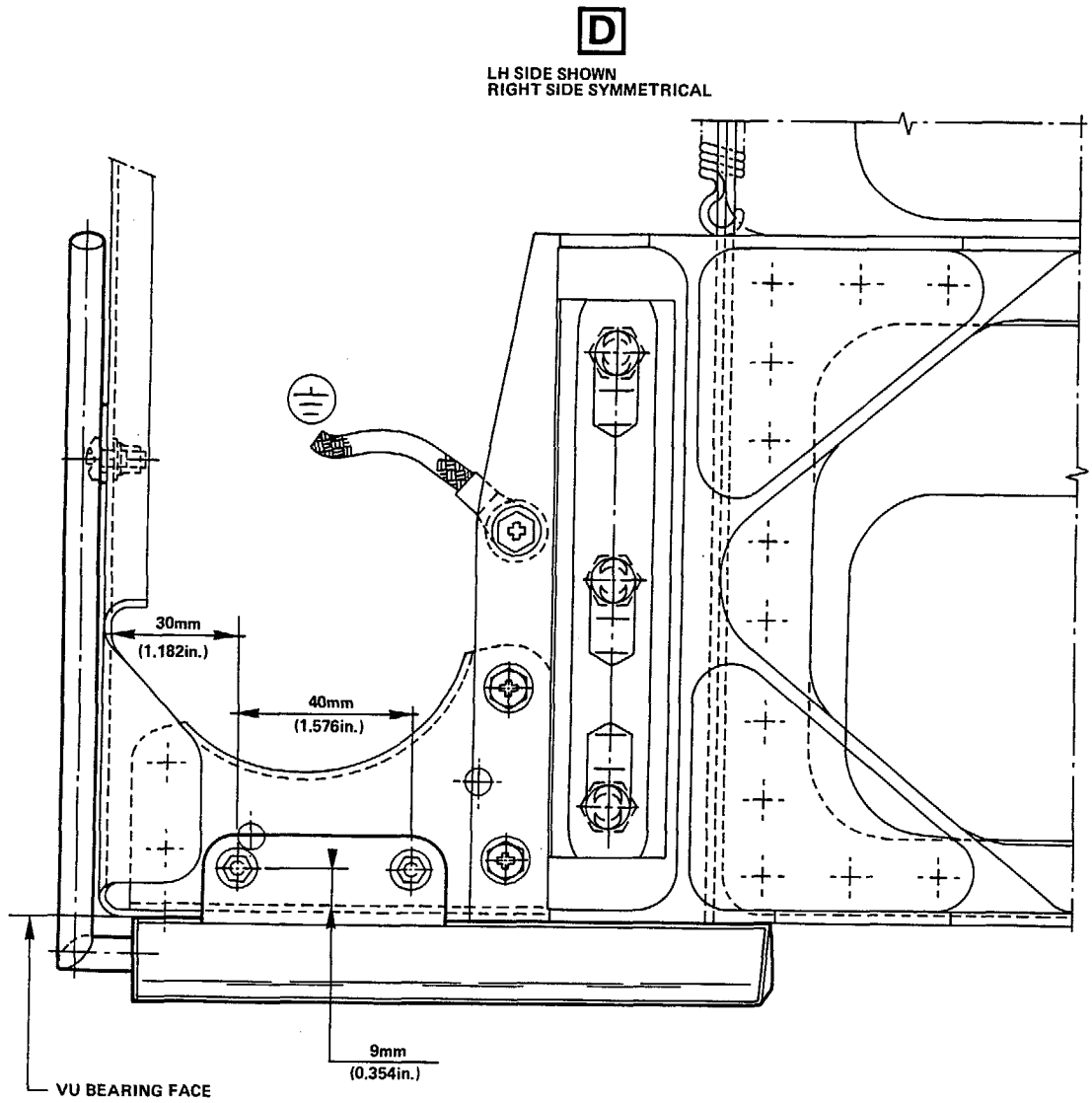
Printed in France





Installation of a Trough under the Lower Post of the Front Window Frame at 301VU and 500VU
Figure 5, Sheet 2

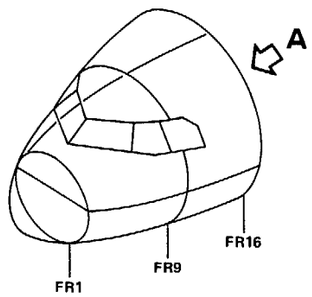
Printed in France



NSB5 211072 AF 00

Installation of a Trough under the Lower Post of the Front Window Frame at 301VU and 500VU
Figure 5, Sheet 3

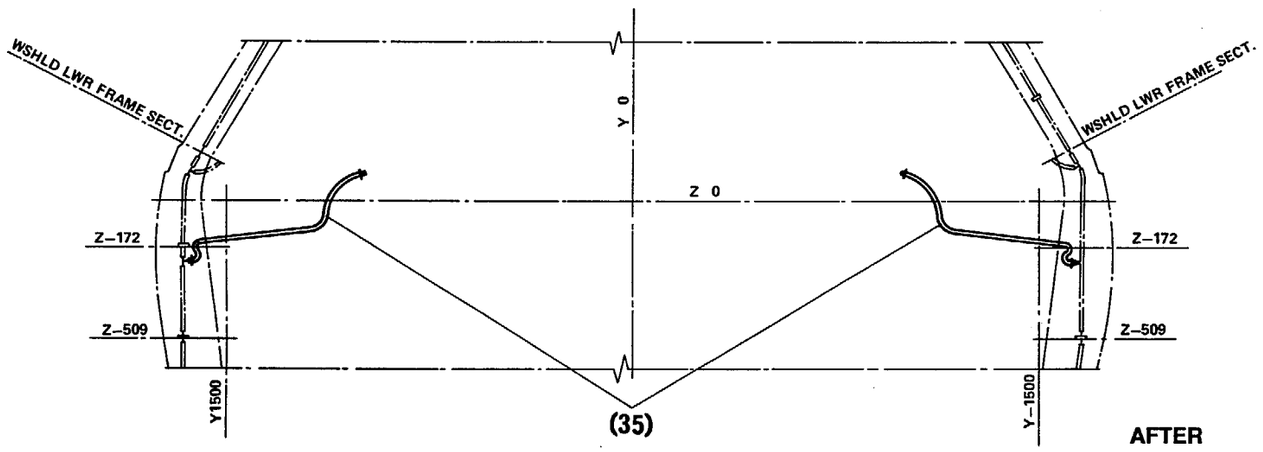
Printed in France



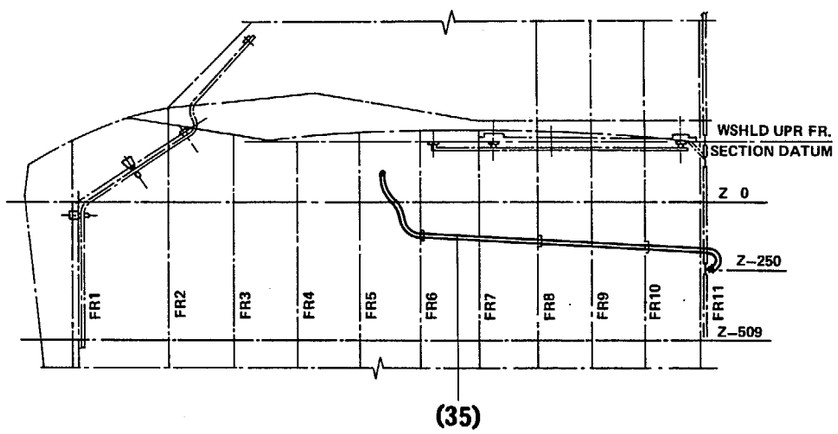
A

BEFORE

VIEW LOOKING FWD FROM FR16 TO FR18

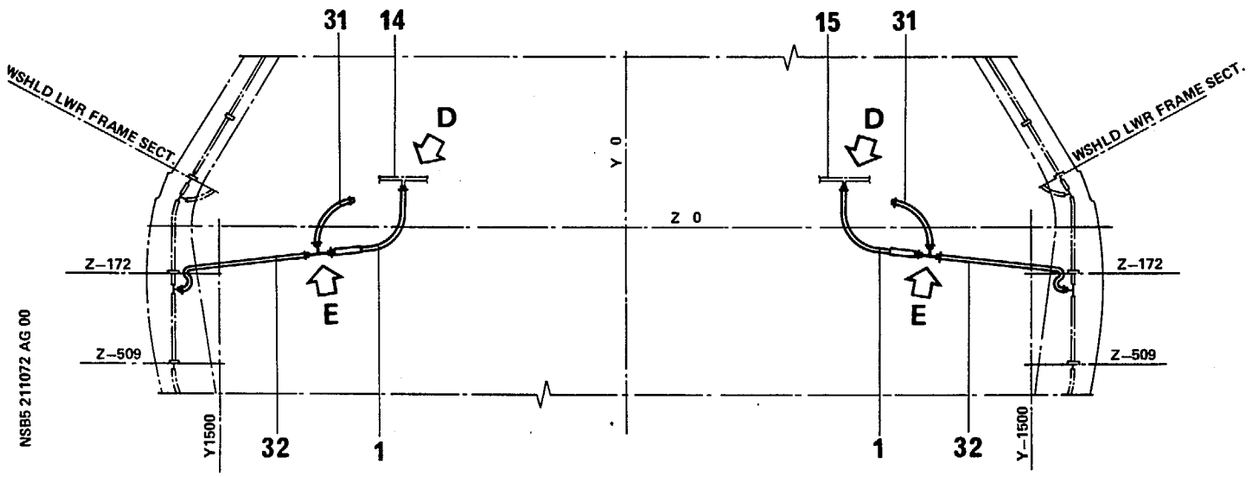


LEFT SIDE VIEW

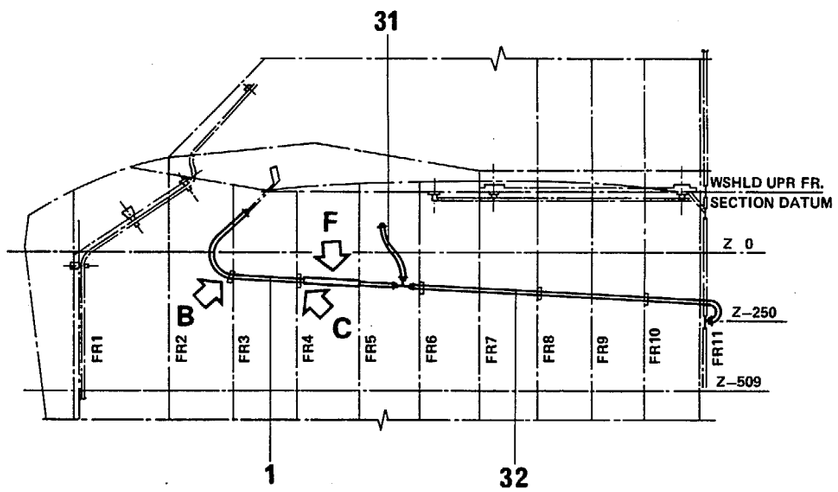


AFTER

VIEW LOOKING FWD FROM FR16 TO FR18

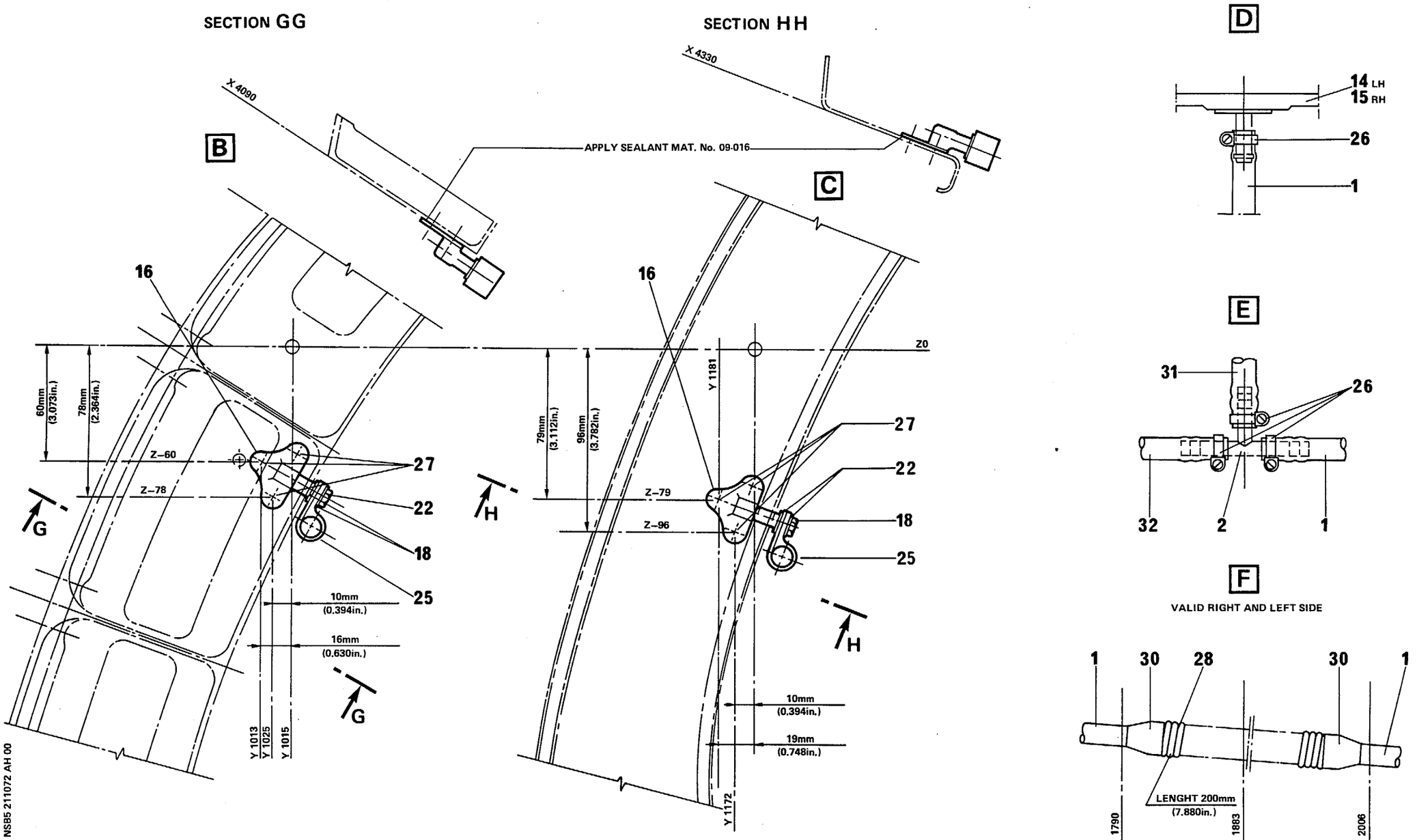


LEFT SIDE VIEW



Modification of Drain Line under L and R Lower Windshield Front Section
Figure 6, Sheet 1 of 2

Printed in France

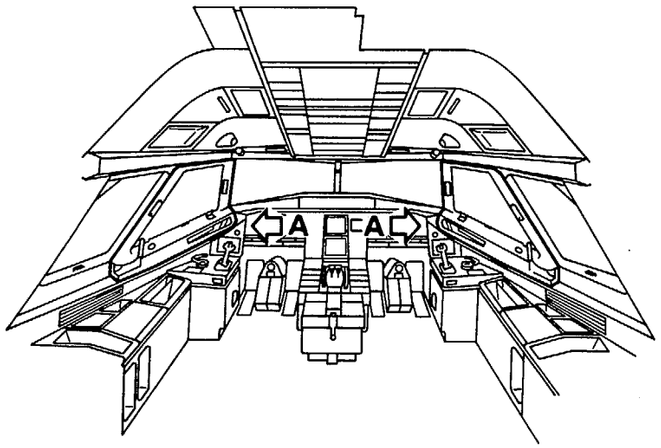


F

VALID RIGHT AND LEFT SIDE

Modification of Drain Line under L and R Lower Windshield Front Section
Figure 6, Sheet 2

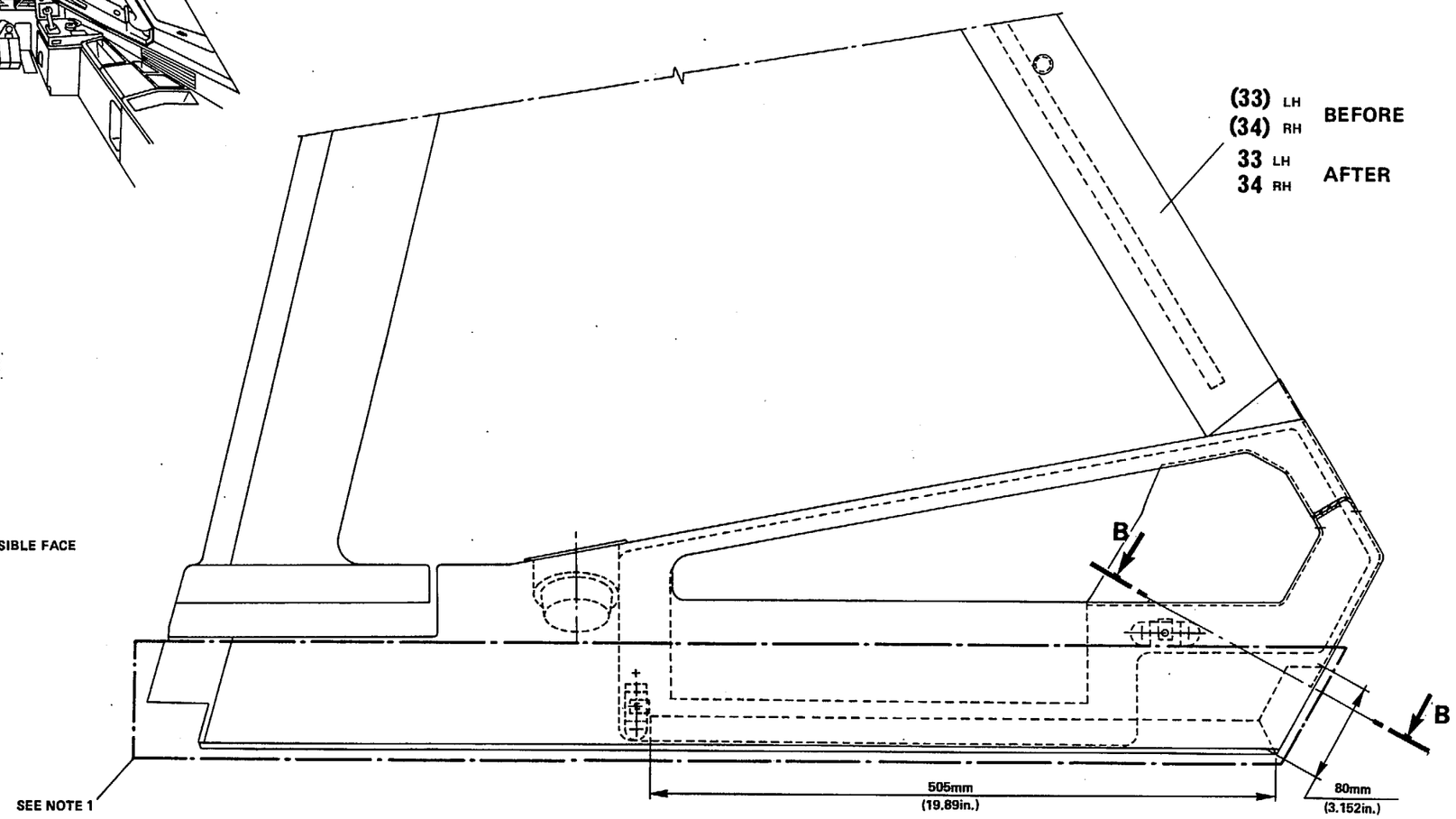
Printed in France



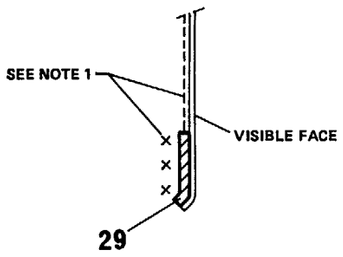
NOTE 1 : ON VISIBLE FACE + INSIDE AREA – – –
– – – PAINTED AREA THICKNESS 0.130mm (0,005in.)
× × × PROTECTED AREA (SEE SECTION BB)

A

LH SIDE SHOWN
RH SIDE SYMMETRICAL



SECTION B B



NSB5 211072 AJ 00

Installation of the Seal on the Lower Part of the Sliding Window Easings
Figure 7

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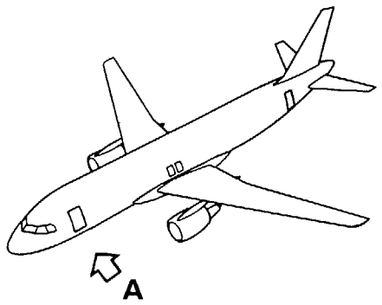

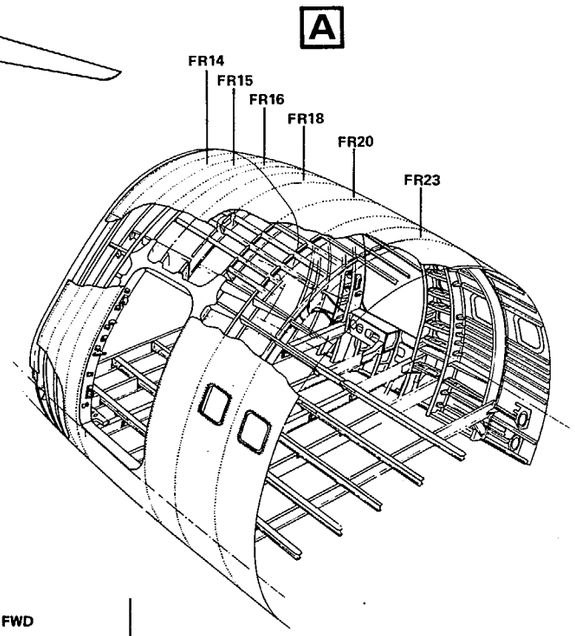

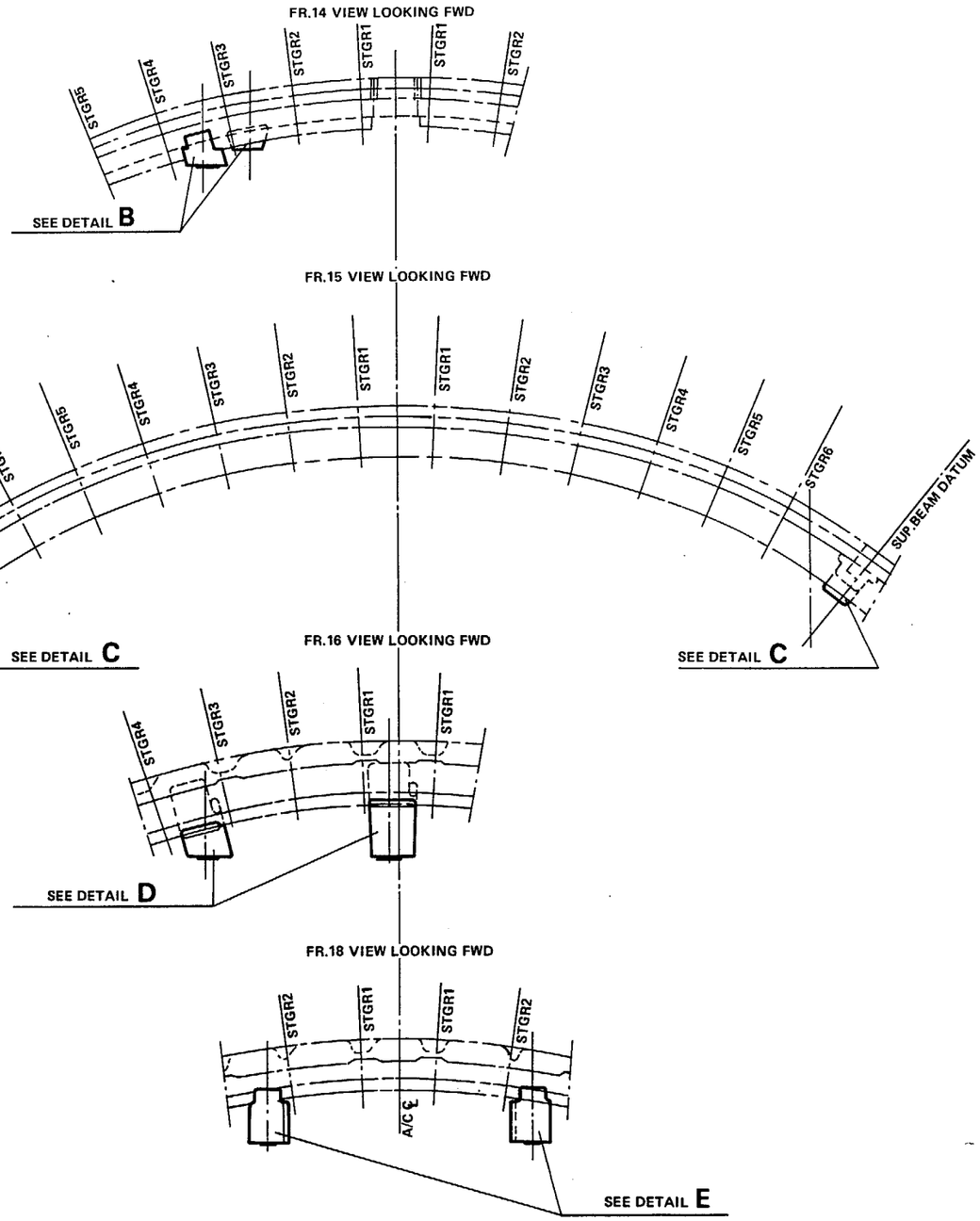

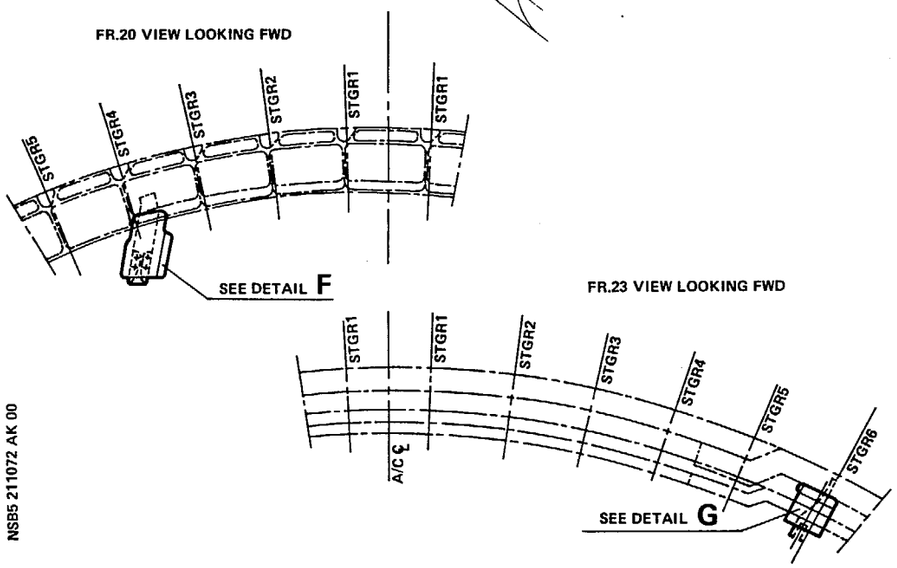

Installation of the Insulation Blankets on the Visible Structures
Figure 8, Sheet 1 of 7

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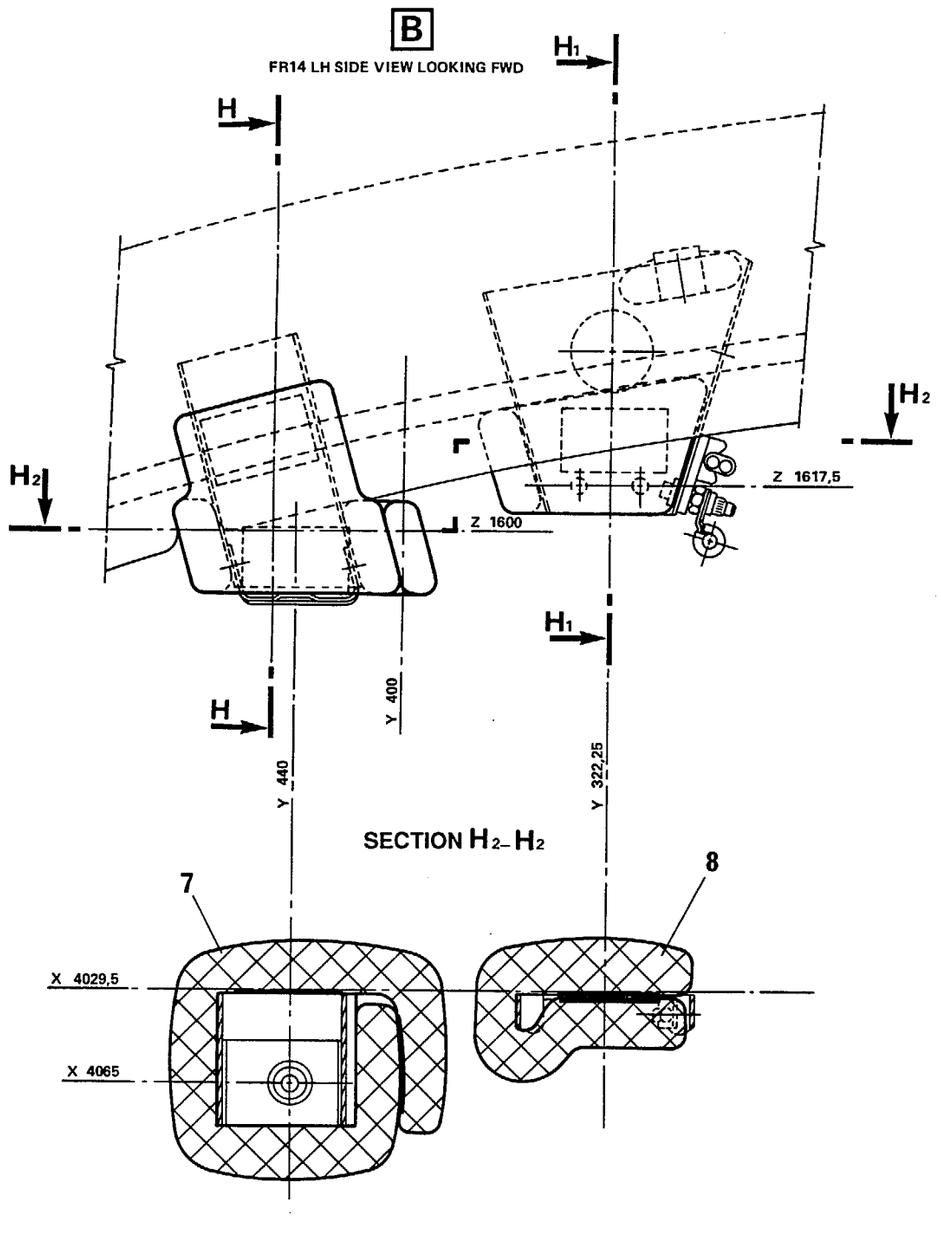

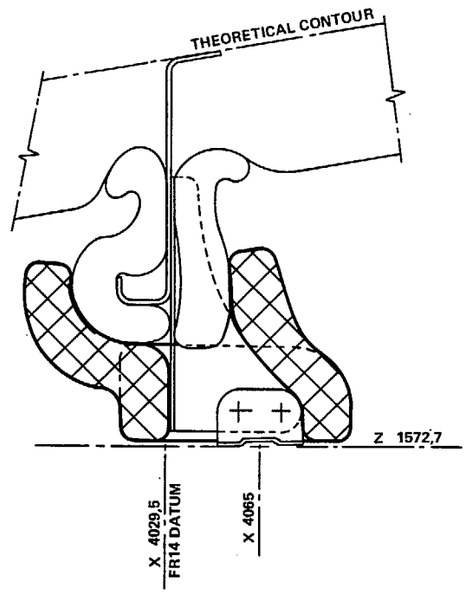

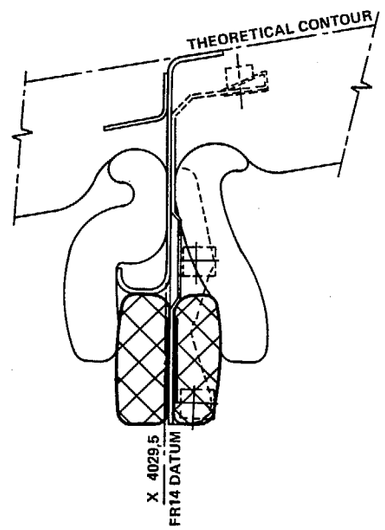

Installation of the Insulation Blankets on the Visible Structures
Figure 8, Sheet 2

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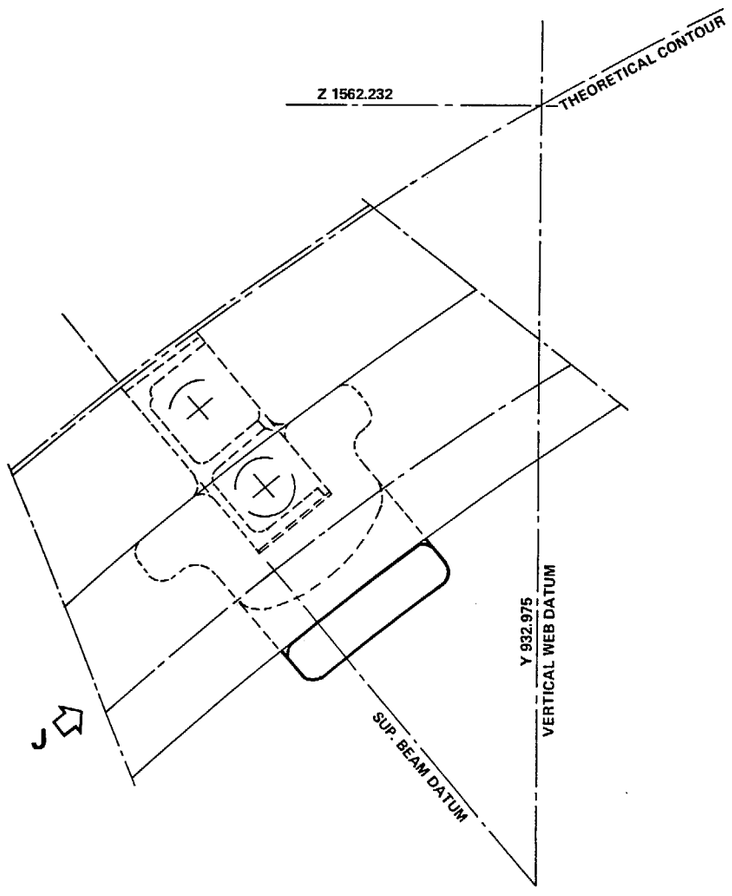
C

FR15 LH SIDE VIEW LOOKING FWD , RH SIDE SYM.



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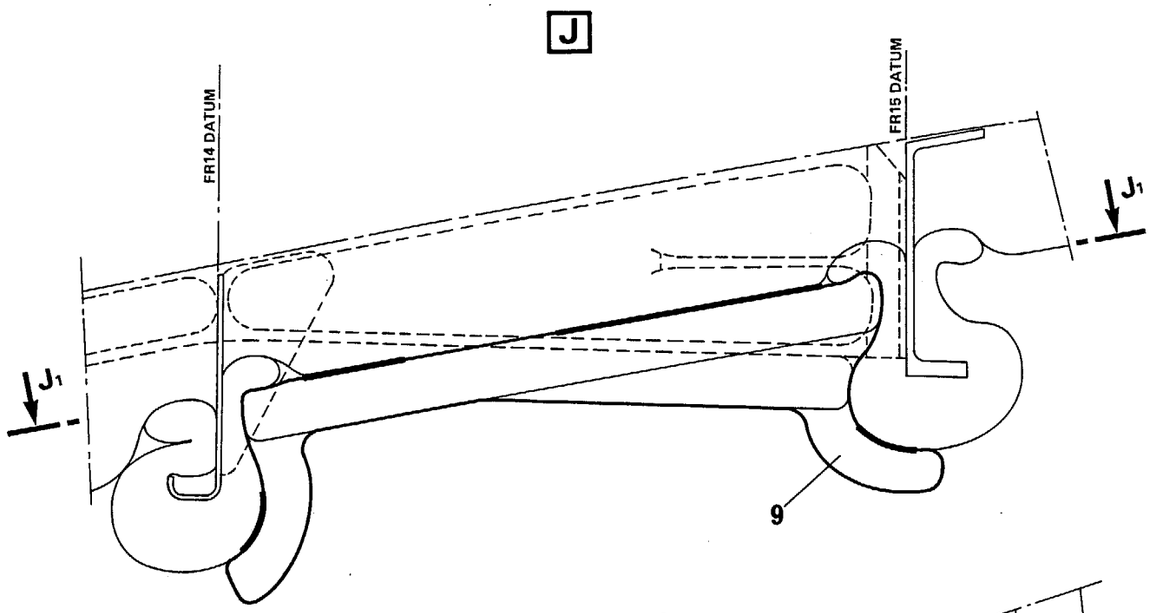
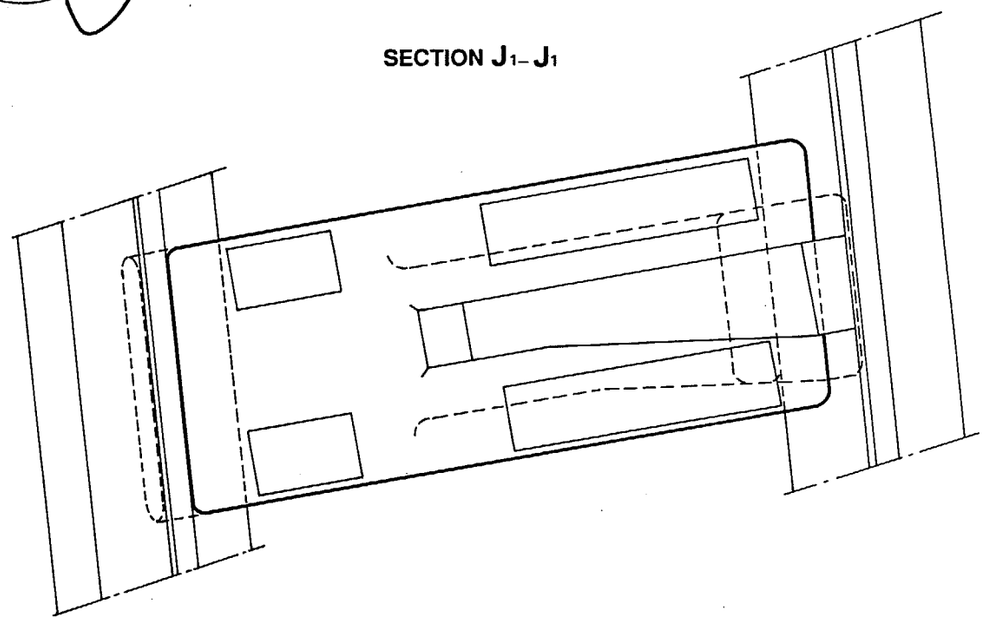


SECTION J1– J1



NSB5 211072 AM 00

Installation of the Insulation Blankets on the Visible Structures
Figure 8, Sheet 3

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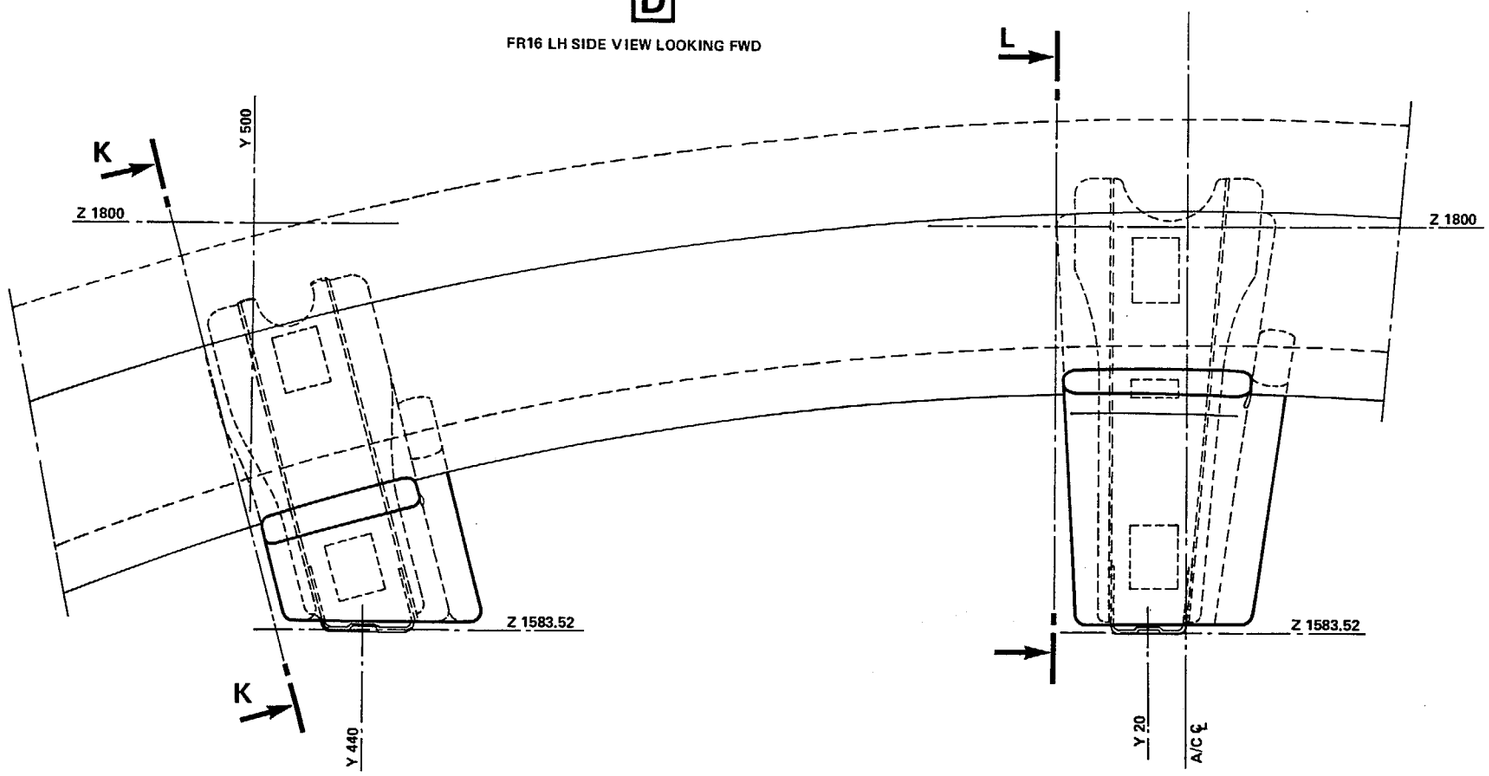
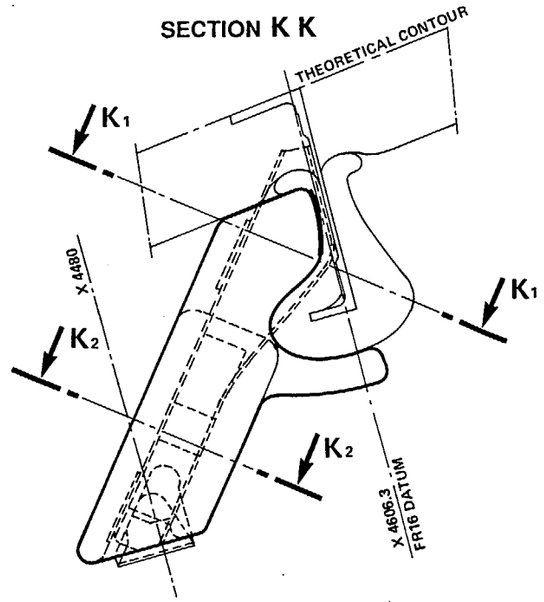
D

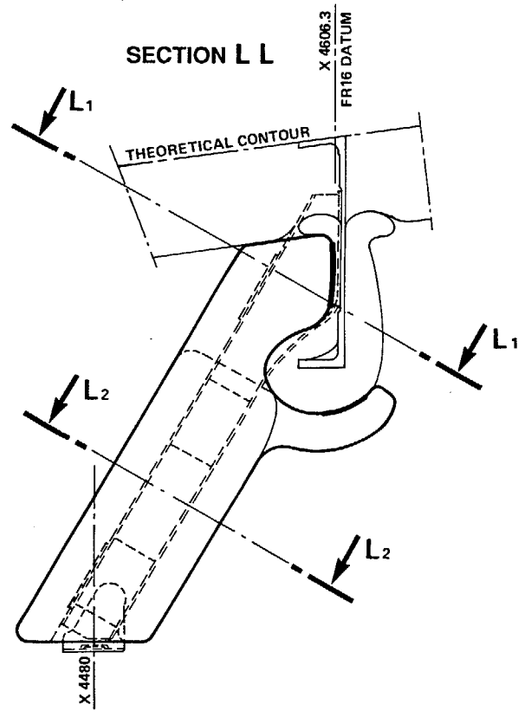
FR16 LH SIDE VIEW LOOKING FWD



SECTION K K



SECTION L L



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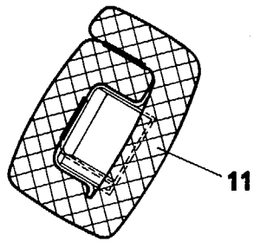

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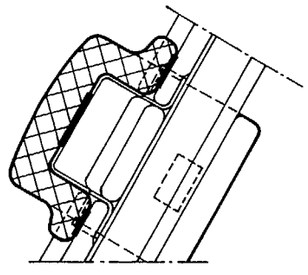

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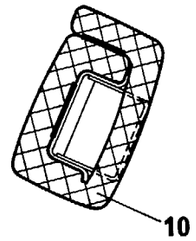
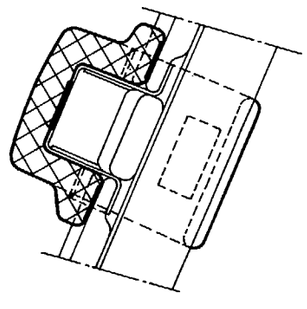


SECTION K1-K1



NSB5 211072 AN 00

Installation of the Insulation Blankets on the Visible Structures
Figure 8, Sheet 4

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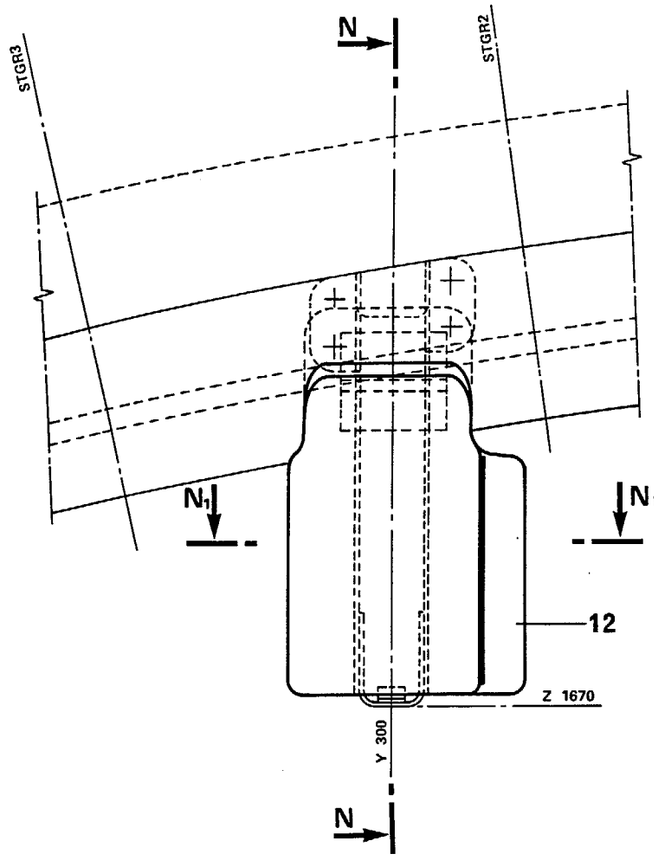
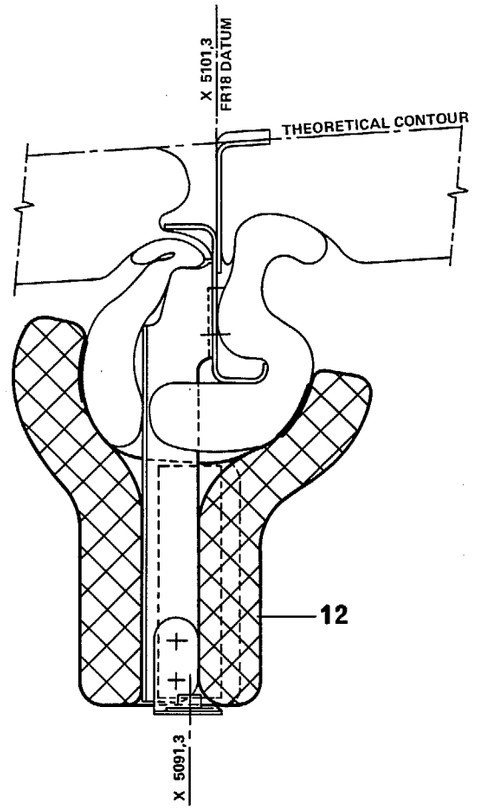
E

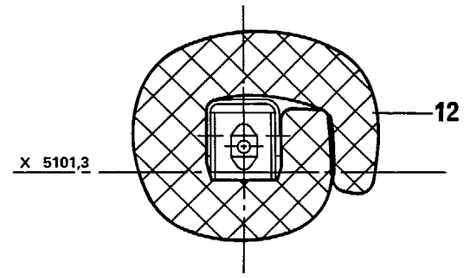
FR18 LH SIDE VIEW LOOKING FWD, RH SIDE SYM.



SECTION N N



SECTION N1- N1



NSB5 211072 AO 00

Installation of the Insulation Blankets on the Visible Structures
Figure 8, Sheet 5

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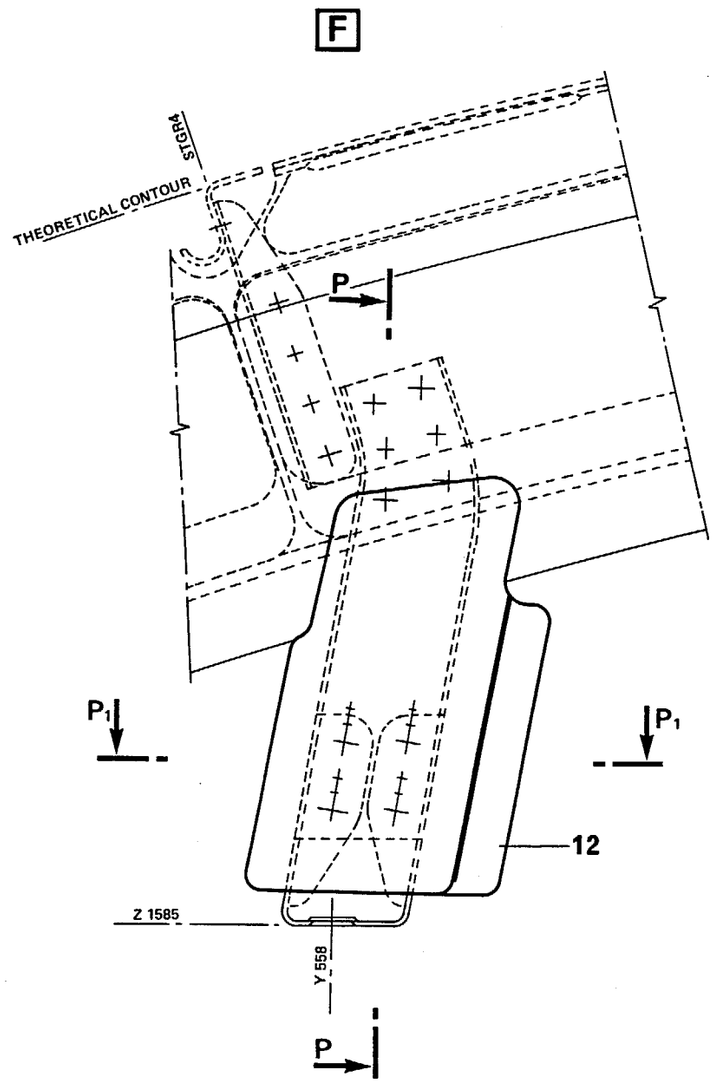
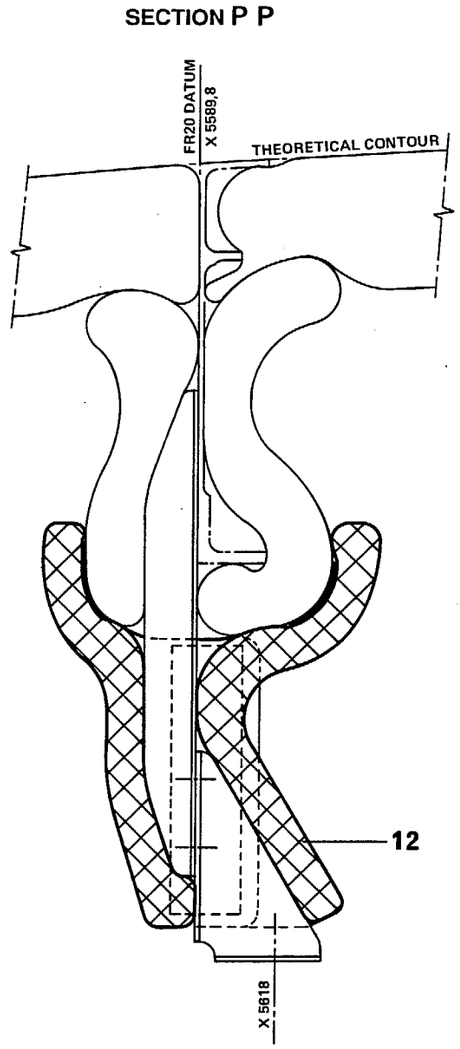
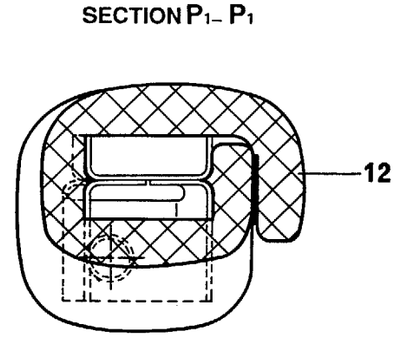


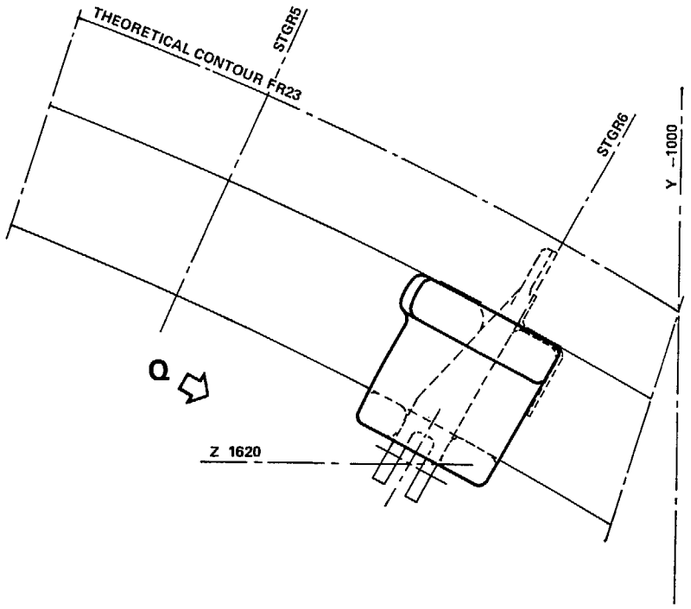


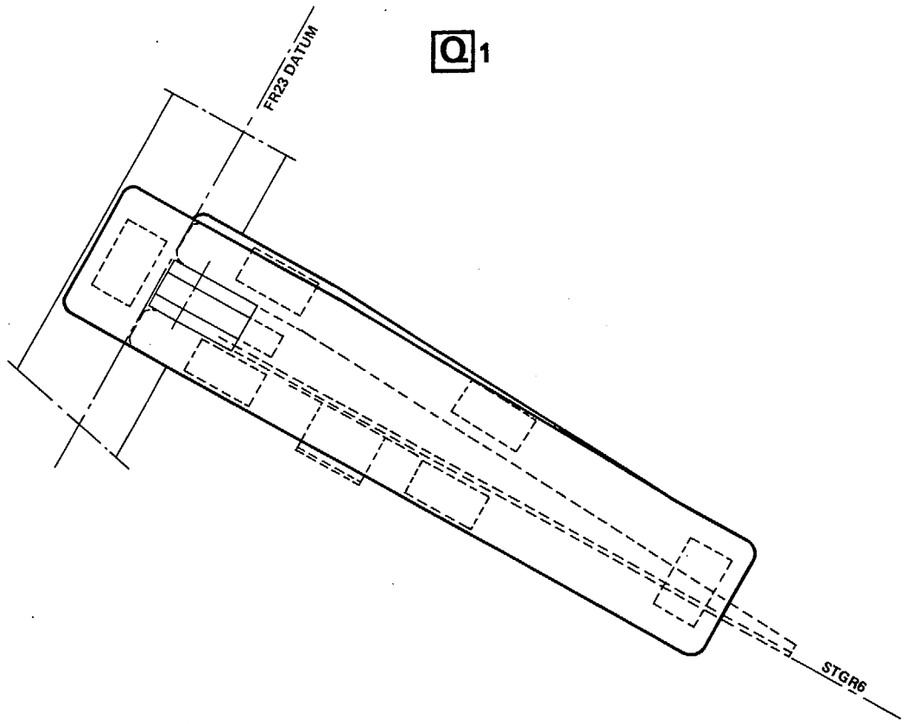


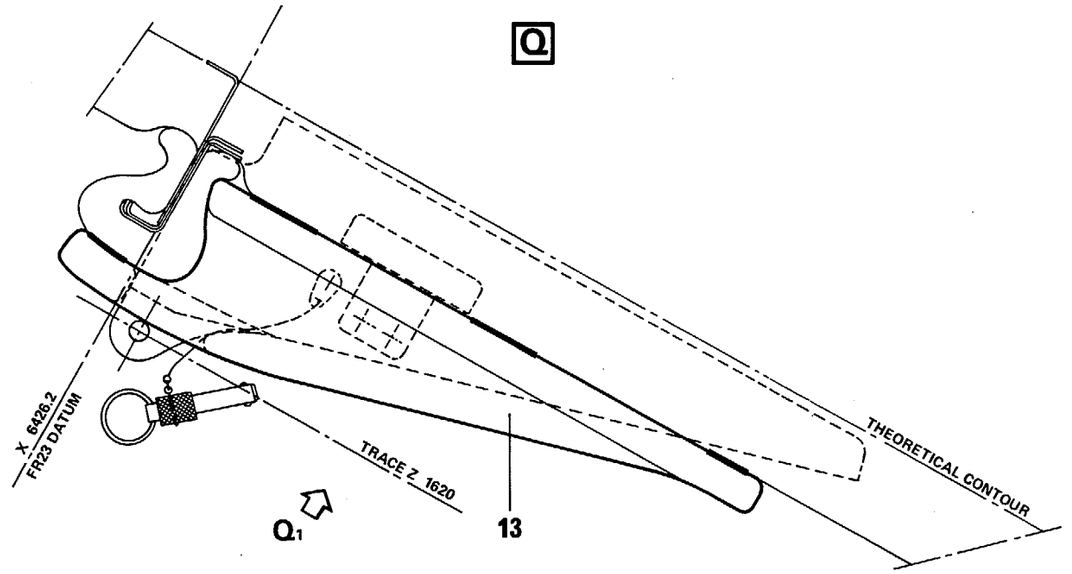
Installation of the Insulation Blankets on the Visible Structures
Figure 8, Sheet 6

DETAIL G

FR23 RH SIDE VIEW LOOKING FWD



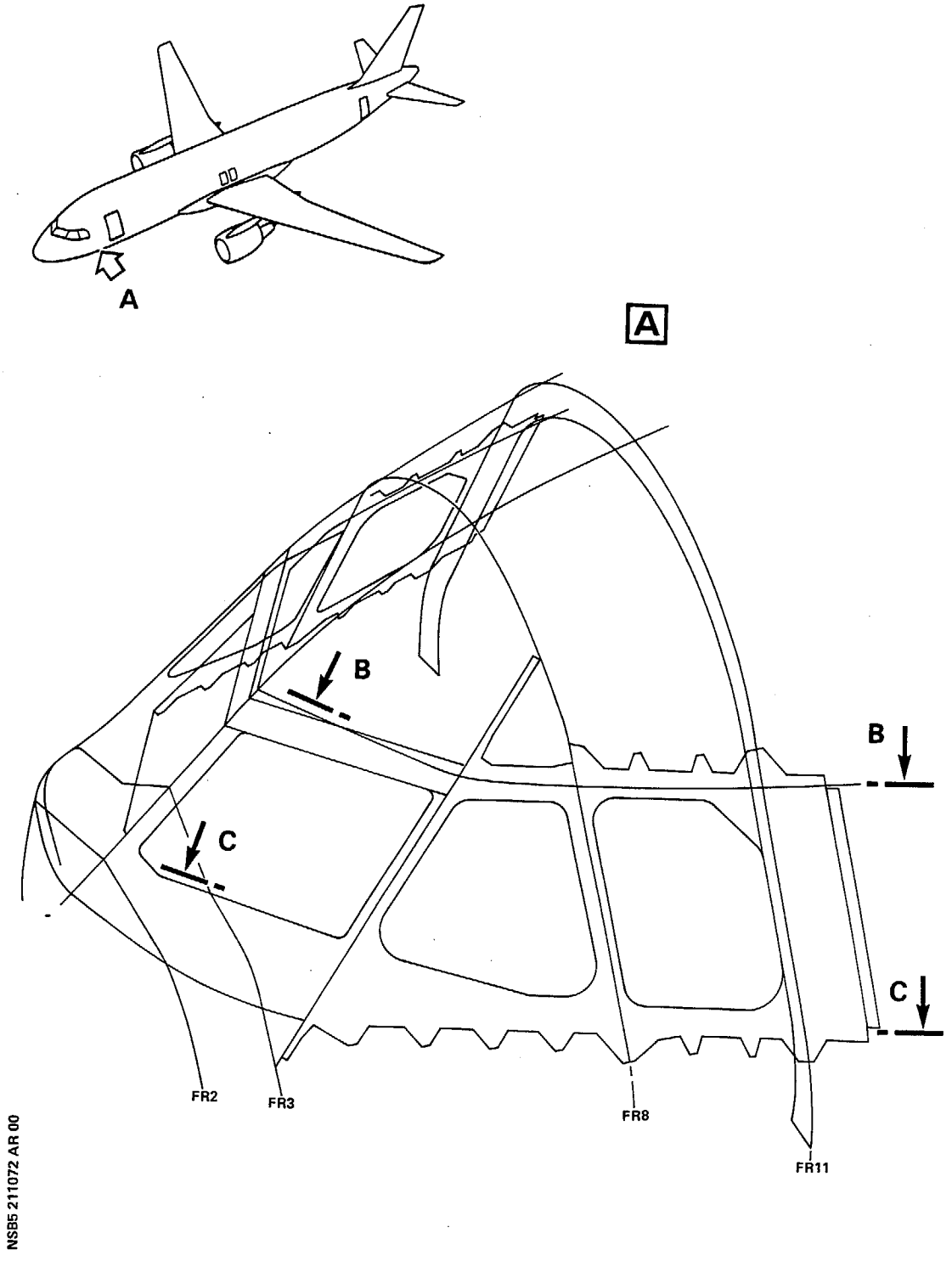




NSB5 211072 AQ 00

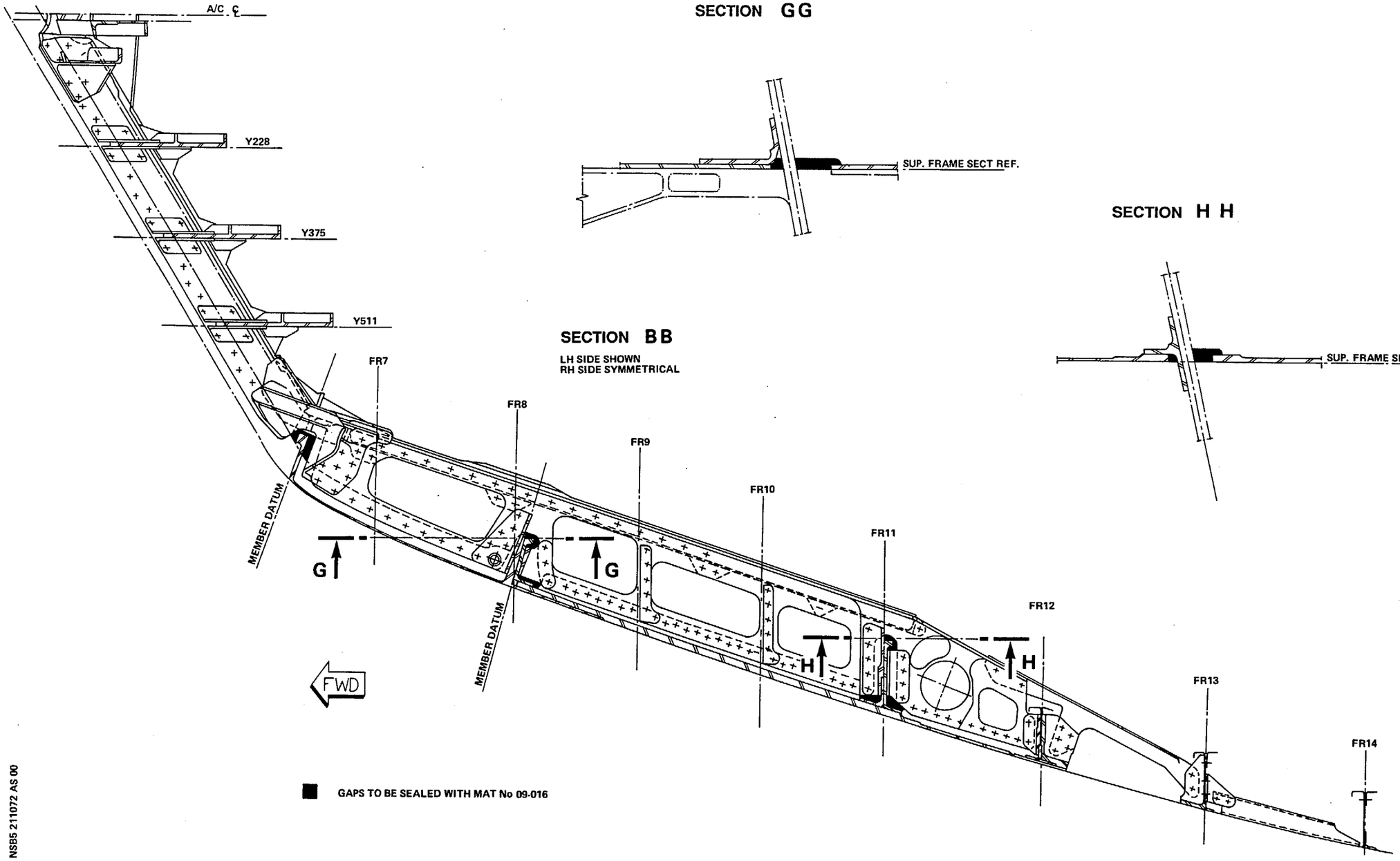
Installation of the Insulation Blankets on the Visible Structures
Figure 8, Sheet 7

Printed in France



Blanking of Structural Holes at the Lower and upper
L and R Windshield front Sections
Figure 9, Sheet 1 of 3

Printed in France



SECTION G G

SECTION B B

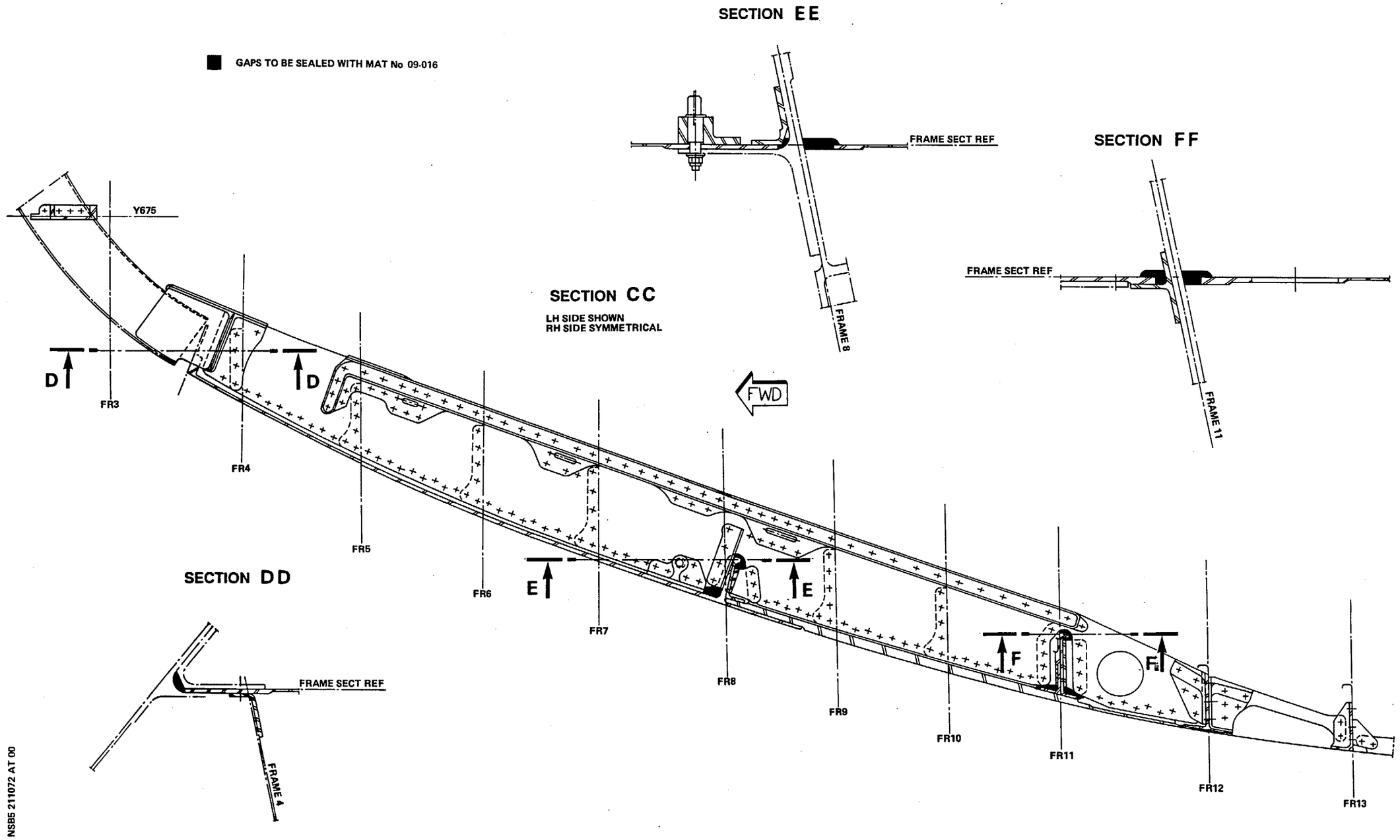
LH SIDE SHOWN
RH SIDE SYMMETRICAL

SECTION H H

NSB5 211072 AS 00

Blanking of Structural Holes at the Lower and Upper
L and R Windshield front Sections
Figure 9, Sheet 2

Printed in France



Blanking of Structural Holes at the Lower and Upper
L and R Windshield front Sections
Figure 9, Sheet 3

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3. MATERIAL INFORMATION

A. LIST OF COMPONENTS

ITEM	NEW PART No.	QTY	DESCRIPTION	ITEM	OLD PART No.	INT	INSTRUC	DISPOS.
	Kit No. 211072	A01						
1	A2181060426800	2	Pipe					
2	D0003002400100	2	Tee					
3	D2511157000000	1	Panel assy	(3)	D2511056300000	03	*	
4	D2511157000100	1	Panel assy	(4)	D2511056300100	03	*	
5	D2511158100000	1	Cover assy	(5)	D2511085100400	03	*	
6	D2511158100100	1	Cover assy	(6)	D2511085100500	03	*	
7	D2581053400000	1	Blanket-Insulation					
8	D2581053500000	1	Blanket-Insulation					
9	D2581053600000	2	Blanket-Insulation					
10	D2581053700000	1	Blanket-Insulation					
11	D2581053800000	1	Blanket-Insulation					
12	D2581053900000	3	Blanket-Insulation					
13	D2581054000000	1	Blanket-Insulation					
14	D9251277800000	1	Trough assy					
15	D9251277800100	1	Trough assy					
16	ASNA2328-3-4	4	Bracket					
17	NAS1096-1-5	4	Screw					
18	NAS1096-3-10	4	Screw					
19	NAS1100-08-10	2	Screw					
20	ASNA2397-6L	4	Washer					
21	ASNA2397-8	2	Washer					
22	ASNA2397-10L	8	Washer					
23	MS21042-08	2	Nut					
24	MS21042-06	4	Nut					
25	NSA5515-7	4	Clamp					
26	ASNA0033-022	8	Clamp					
27	ASNA2050DCJ3215	12	Rivet					

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ITEM	NEW PART No.	QTY	DESCRIPTION	ITEM	OLD PART No.	INT	INSTRUC	DISPOS.
	Kit No. 211072	A01						
28	E0432A16	200mm (7.88in.)	Conduit					
29	ABS5006-4D	200mm (7.88in.)	Strip-Silicone					
30	ASNA35622420	1	Tape-Adhesive					

B. SPECIAL TOOLS

None.

C. LIST OF MATERIALS - OPERATOR SUPPLIED

DESCRIPTION	REFERENCE CML	QTY PER A/C
PR1436GB2	Mat. No. 09-016	As required
MEK	Mat. No. 11-003	As required
Epoxy primer	Mat. No. 16-001	As required
Polyurethane grey	Mat. No. 16-002	As required
Decorative paint	Mat. No. 16-047	As required

D. PARTS TO BE RE-IDENTIFIED BY OPERATOR

ITEM	NEW PART No.	DESCRIPTION	ITEM	OLD PART No.
31	A2181060420200	Pipe	(35)	D2121069821200
32	D2121069821100	Pipe		
33	D2511158400000	Carter assy	(33)	D2511105300200
34	D2511158400100	Carter assy	(34)	D2511105300300

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MODIFICATION No. 23554P3127 ATA SYSTEM : 21

TITLE : AIR CONDITIONING - COCKPIT - INTRODUCE PROTECTION AGAINST WATER CONDENSATION.

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