

See discussions, stats, and author profiles for this publication at: <https://www.researchgate.net/publication/339344112>

Surface temperature measurement of gas turbine combustor using temperature-indicating paint

Article in *International Journal of Ambient Energy* · February 2020

DOI: 10.1080/01430750.2020.1731709

CITATIONS

27

READS

890

2 authors, including:



Rupesh Kumar P L

Vel Tech Rangarajan Dr. Sagunthala R&D Institute of Science and Technology

26 PUBLICATIONS 123 CITATIONS

SEE PROFILE



Surface temperature measurement of gas turbine combustor using temperature-indicating paint

M. Arulprakasajothi & P. L. Rupesh

To cite this article: M. Arulprakasajothi & P. L. Rupesh (2020): Surface temperature measurement of gas turbine combustor using temperature-indicating paint, International Journal of Ambient Energy, DOI: [10.1080/01430750.2020.1731709](https://doi.org/10.1080/01430750.2020.1731709)

To link to this article: <https://doi.org/10.1080/01430750.2020.1731709>



Accepted author version posted online: 18 Feb 2020.
Published online: 04 Mar 2020.



Submit your article to this journal [↗](#)



View related articles [↗](#)



View Crossmark data [↗](#)



Surface temperature measurement of gas turbine combustor using temperature-indicating paint

M. Arulprakasajothi and P. L. Rupesh

Department of Mechanical engineering, Veltech Rangarajan Dr Sagunthala R&D Institute of Technology, New Vellanur, India

ABSTRACT

Surface temperature measurement of a hot surface is necessary in order to determine its thermal life. The combustors used in gas turbine systems may be subject to high thermal load due to continuous combustion. In order to analyse the surface temperature distribution of combustor, temperature-sensing instruments are necessary. Temperature-indicating paint is a kind of coating to measure the temperature distribution through its colour change. Temperature-indicating paint is considered to be one of the most efficient ways for temperature measurement. The analysis of this paint can be done through the observation of irreversible colour change with temperature. This work deals with the fabrication of gas turbine combustor, investigates the thermal distribution and finds the hot spots of the combustor using temperature-indicating paint.

ARTICLE HISTORY

Received 7 January 2020
Accepted 30 January 2020

KEYWORDS

Temperature-sensitive paints; temperature measurement; Thermocouples; Combustor

1. Introduction

An internal combustion engine in which continuous combustion takes place is called gas turbine (or) combustion turbine. The system consists of a compressor, a turbine and a combustion chamber. The Brayton cycle is the air standard cycle for the operation of a gas turbine engine (Lempereur, Andral, and Prudhomme 2008; Bird et al. 1998). The compressed atmospheric air from the compressor is sent to the combustion chamber where the fuel is injected (or) sprayed. The injection of fuel with the high pressurised air leads to combustion which produces high pressurised gases (Neely and Riesen 2008; Bird et al. 1998). These gases with high pressure undergo expansion in the turbine which produces work. An industrial gas turbine with the above parts has been shown in Figure 1.

2. Combustor

Figure 2 depicts the shape of the combustor along with its accessories. The combustor plays a vital role in the operation of a gas turbine engine. The combustible gases produced may increase the heat load on the walls of the liner of the combustion chamber. The combustion chamber is produced with different slots (or) paths for air circulation to decrease the heat load. (Dolvin 2008; Griffin et al. 1996; Iliff and Shafer 1993; Jones and Hunt 1964). The air flows into the combustion chamber splits itself into three types as primary, secondary and tertiary to improve the combustion and for flame stabilisation, as shown in Figure 3 (Lee, Dutton, and Jacobi 2007).

3. Combustor-design and fabrication

A combustor has been designed and fabricated, as shown in Figures 4–5, with the dimensions given in Table 1.

4. Selection of paint

MC 350–8 which is of multi-change-type thermal paint has been selected for temperature detection. The paint's initial colour is red and the colour change starts to appear only after 350°C and exhibits about 8 different colours at different elevated temperatures (Padgham and Saunders 1975), as mentioned in the datasheet. The paint consists of a resin and a pigment which is responsible for colour change (Connolly 2009; Cowling, King, and Alexander 1953). The resin is a solvent-based acrylic and silicon of composition (solids-59.5%, VOC-635 g/l, flash point 30°C) (Kojima et al. 2004; Yang and Zhi-Min 2014). The pigment consists of C.I pigment Red 104, lead chromate. The paint may be thickened if required by the addition of polymethyl acrylate. The standard colour changes have been observed when a metal coupon is coated with MC 350-8 on exposure to continuous heating as soon as the temperature is more than 350°C (Mandavkar et al. 2008) which is depicted in Figure 6.

5. Spraying and drying of thermal paint on specimens

The combustor was sprayed with thermal paint according to the following steps (Liu et al. 1997; Liu and Sullivan 2004), as shown in Figure 7.

- The surface treatment is done by polishing the metal surface using a sandpaper so that the uneven surface regions and rust can be removed.
- The polished combustor is cleaned by dipping in acetone for about 15 min so that chemical contaminants are removed.
- After the surface cleaning the combustor is heated in the furnace to 200°C to burn out the oil traces in the specimen.
- The paint is sprayed on the surface of the specimen.

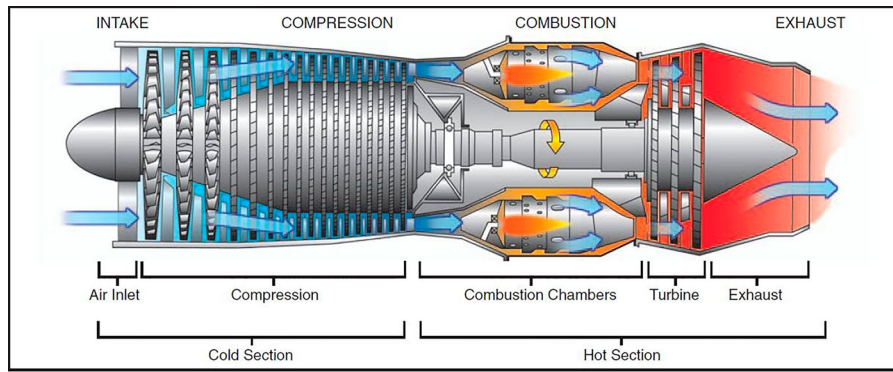


Figure 1. Industrial gas turbine.

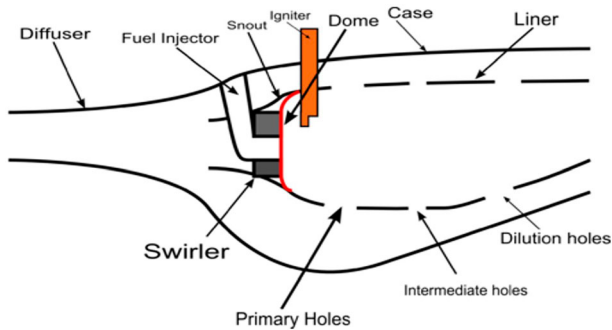


Figure 2. Components of a combustor.

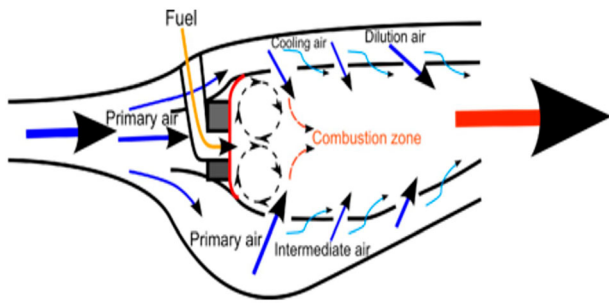


Figure 3. Airflow paths of a gas turbine.

- The combustor is again heated for a curing temperature of about 250°C for about 2 h and cools to room temperature in the oven itself Figure 8.

6. Experimental test rig

The schematic diagram represents the actual set-up of the test rig in a schematic way. It shows how the circuit is working. Compressed air is flowed using the compressor into the casing to help the combustion process. The fuel is flowed by the gas torch nozzle into the combustor. The igniters produce a spark and start the ignition process in the combustor. Flame starts to flow inside the combustor and thermal distribution occurs in the surface.

The experimental test rig is constructed with the help of the components described above, as it is shown in Figure 9. The top view of the set-up is shown in Figure 9(a) and coupled with a

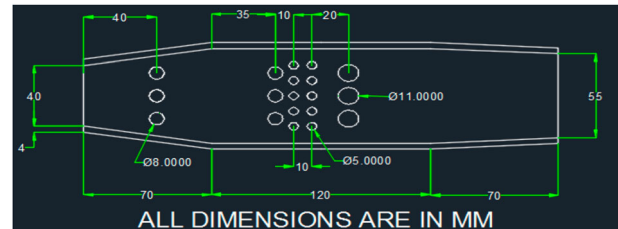


Figure 4. 2D modelling of a combustor.



Figure 5. Fabrication of a combustor.

Table 1. Specification of a combustor.

Material used	Stainless steel 304
Melting point	1535°C
Length	160 mm
Thickness	4 mm
Primary hole diameter	7 mm
Secondary hole diameter	5 mm
Dilution hole diameter	10 mm
Primary hole distance from one end	105 mm
Secondary hole distance from one end	115 mm
Dilution hole distance from one end	145 mm
No of primary hole	16
No of secondary hole	16
No of dilution hole	6

compressor also shown in the side view of the set-up is shown in Figure 9(b).

7. Results and discussion

Figure 10 shows the temperature distribution in the combustor after combustion (Dolvin 2008) and temperature is mentioned by using the calibrated metal coupons. This colour in the flame tube is obtained after 300 min of combustion (Bown et al. 1994) Figure 11.

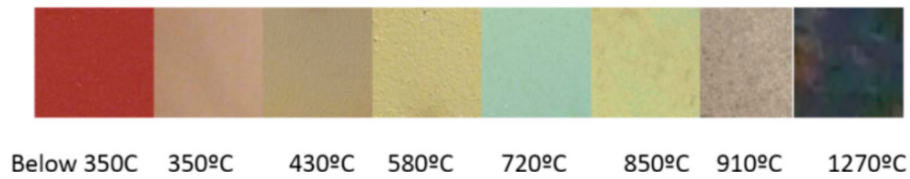


Figure 6. Standard colour changes observed at different temperatures from datasheet.

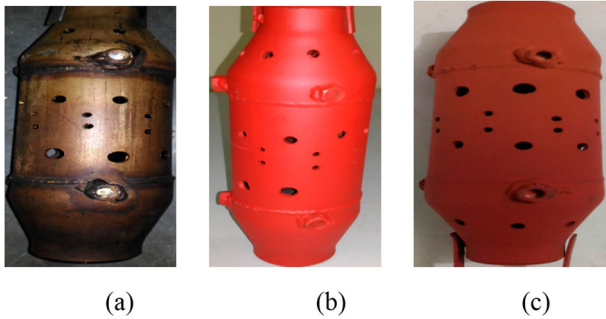


Figure 7. Combustor (a) Acetone dipping (b) After painting (c) After curing.

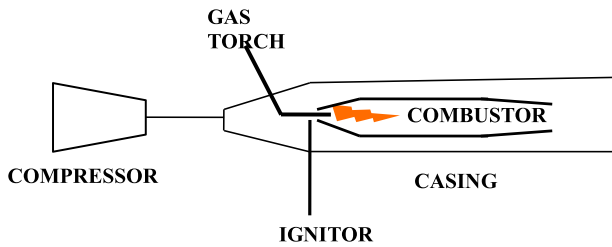


Figure 8. Schematic diagram of the experimental test rig.

8. Hot spot identification in flame tube and combustor

Hot spot is nothing but unpredictable temperature difference in the surface, material defect, improper composition, improper



Figure 10. Image after combustion of 300 min.

surface, etc . . . (Neely and Tracy 2006). These may be reasons behind the hot spot in the flame tube. But the temperature-indicating paint shows the thermal distribution of the flame tube very accurately in the pixel also. The hot spot in the flame tube after combustion is shown in Figure 12.

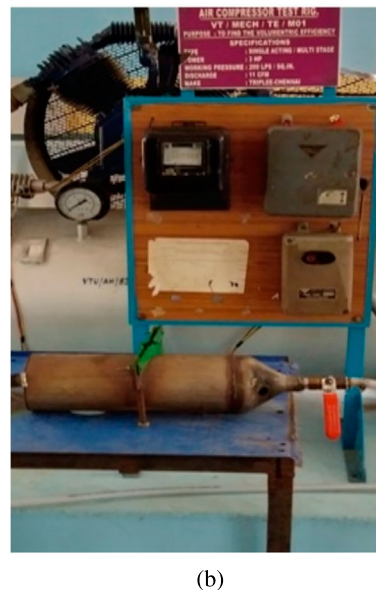


Figure 9. Experimental set-up (a) Top view (b) Side view.

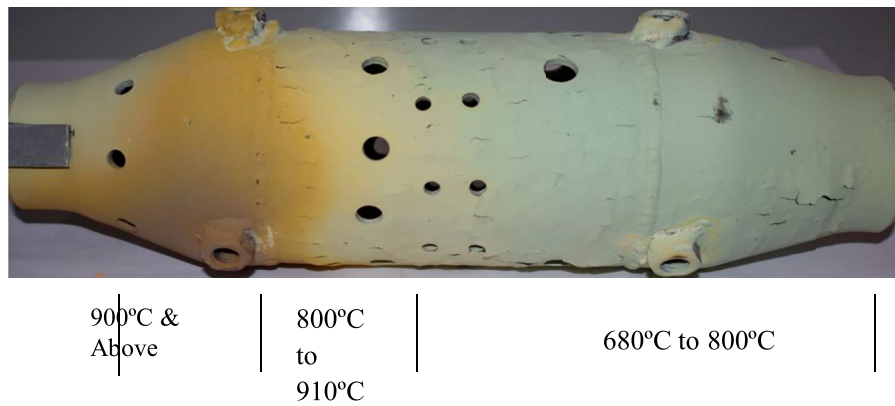


Figure 11. Temperature distributions in a combustor.

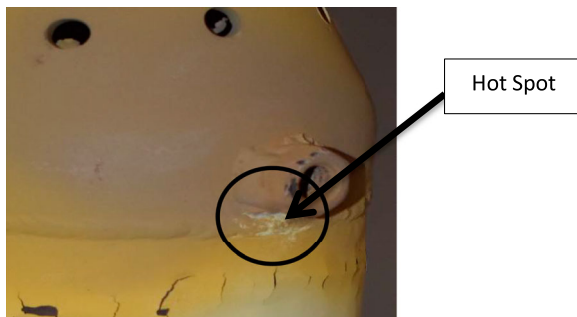


Figure 12. Hotspots in a combustor.

9. Conclusion

The high pressurised air from the compressor is decelerated through the diffuser and is distributed as primary air through the holes present in the dome of the combustion chamber. The primary air helps in starting combustion. Some of the pressurised air is bypassed and passes through the secondary holes to complete the reaction process to dilute carbon monoxide (CO) and hydrocarbon (HC) emissions. The dilution air helps in reducing the temperature of hot gases released by the combustion. This dilution makes the turbine blades to withstand the temperature of combustible gases within the permissible limit. The results and discussions observed from the above work reveal that temperature-indicating paint is affordable technique to evaluate the temperature distribution of a surface which is subjected to high temperature. MC 350-8 the thermal paint which is used in the above work has proved successful in providing the temperature distribution with distinguishable colour contours.

Disclosure statement

No potential conflict of interest was reported by the author(s).

References

- Bird, C., J. E. Mutton, R. Shepherd, and M. D. W. Smith. 1998. "Surface Temperature Measurement in Turbines." AGARD conference proceedings 598, Pg. No: 21.421.10.
- Bird, C., J. E. Mutton, R. Shepherd, M. D. W. Smith, and H. M. L. Watson. 1998. "Surface Temperature Measurement in Turbines", AGARD-CP-598.
- Bown, N. W., T. M. Cain, T. V. Jones, P. P. Shipley, and B. Barry. 1994. "In Flight Heat Transfer Measurements on an Aero-Engine Nacelle", ASME 94-GT-244.
- Connolly, M. 2009. Temperature Measurement Using Thermal Paints, Rolls-Royce Deutschland, Germany.
- Cowling, J. E., P. King, and A. L. Alexander. 1953. "Temperature-Indicating Paints." *Industrial and Engineering Chemistry* 45 (10): 2317–2320.
- Dolvin, D. J. 2008. "Hypersonic International Flight Research and Experimentation (HIFiRE) Fundamental Sciences and Technology Development Strategy", AIAA-2008-2581.
- Griffin, A., J. Kittler, T. Windeatt, and G. Matas. 1996. "Techniques for the Interpretation of Thermal Paint Coated Samples," Proceedings of the 1996 Conference on Pattern Recognition, IEEE, New York, 959–963.
- Iliff, K. W., and M. Shafer. 1993. A Comparison of Hypersonic Vehicle Flight and Prediction Results", AIAA-93-0311.
- Jones, R. A., and J. L. Hunt. 1964. "Use of Temperature-Sensitive Coatings for Obtaining Quantitative Aerodynamic Heattransfer Data." *AIAA Journal* 2 (7): 1354–1356.
- Kojima, F., S. Fukuda, K. Asai, and K. A. Nakakita. 2004. "Identification of Time and Spacial Varying Heat Flux from Surface Measurements based on TSP technology." Proceedings of the SICE Annual Conference.
- Lee, Joosung J., J. Craig Dutton, and Anthony M. Jacobi. 2007. "Application of Temperature-Sensitive Paint for Surface Temperature Measurement in Heat Transfer Enhancement Applications." *Journal of Mechanical Science and Technology* 21 (8): 1254–1257.
- Lempereur, C., R. Andral, and J. Y. Prudhomme. 2008. "Surface Temperature Measurement on Engine Components by Means of Irreversible Thermal Coatings." *IOP Science Measurement Science and Technology* 105501: 4–5.
- Liu, T., B. T. Campbell, S. P. Burns, and P. Sullivan. 1997. "Temperature and Pressure-Sensitive Paints in Aerodynamics." *Applied Mechanics Review* 50: 227–246.
- Liu, T., and J. P. Sullivan. 2004. *Pressure and Temperature Sensitive Paint*. Heidelberg: Springer-Verlag.
- Mandavkar, Pankaj S., Suraj M. Surjuse, Rishabh D. Sawane, and Deepa G. Dongre. 2008. "Study of Thermal Mapping for Health Monitoring of Gas Turbine Blade." *International Journal of Research In Science & Engineering* 1: 1–6.
- Neely, Andrew J., and Hans Riesen. 2008. "Permanent-Change Thermal Paints for Hypersonic Flight-Test." 15th AIAA International Space Planes and Hypersonic systems and Technologies Conference, 2–3, 5–8, 14–15.
- Neely, A. J., and P. J. Tracy. 2006. "Transient Response of Thermal Paints for use on Short-Duration Hypersonic Flight Tests." AIAA, 2006–8000.
- Padgham, C. A., and J. E. Saunders. 1975. *The Perception of Light and Colour*. London: G. Bell.
- Yang, Li, and Li Zhi-Min. 2014. "The Research of Temperature Indicating Paints and Its Application in Aero-Engine Temperature Measurement." *Asia-Pacific International Symposium on Aerospace Technology (APISAT2014)*, *Procedia Engineering* 99: 1154–1155.