

**DEPARTMENT OF TRANSPORTATION  
FEDERAL AVIATION ADMINISTRATION**

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| H5EU<br>HTM<br>Model FJ Sky-Trac<br>19 October 1972 |
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**TYPE CERTIFICATE DATA SHEET NO. H5EU**

This data sheet which is a part of type certificate No. H5EU prescribes conditions and limitations under which the product for which the type certificate was issued meets the airworthiness requirements of the Federal Aviation Regulations.

Type Certificate Holder                      Helicopter Technik Munchen GmbH & Co., Anlagen KG  
8016 Feldkirchen - Wittelsbacher Str. 11  
Western Germany

**I. Model FJ Sky-Trac Helicopter (Normal Category), Approved 19 October 1972**

|                               |  |                         |                         |                 |                  |                 |                    |     |      |      |      |                    |     |      |      |       |         |     |      |      |      |         |     |      |      |       |
|-------------------------------|--|-------------------------|-------------------------|-----------------|------------------|-----------------|--------------------|-----|------|------|------|--------------------|-----|------|------|-------|---------|-----|------|------|------|---------|-----|------|------|-------|
| Engine                        | 1 Franklin 6AS-335-B   |                         |                         |                 |                  |                 |                    |     |      |      |      |                    |     |      |      |       |         |     |      |      |      |         |     |      |      |       |
| Fuel                          | 100/130 min. grade aviation gasoline   |                         |                         |                 |                  |                 |                    |     |      |      |      |                    |     |      |      |       |         |     |      |      |      |         |     |      |      |       |
| Engine limits                 | <table><tr><td></td><td>HP</td><td>RPM</td><td>IN.HG.</td><td>ALT. FT.</td></tr><tr><td>Maximum continuous</td><td>240</td><td>3200</td><td>38.0</td><td>S.L.</td></tr><tr><td>Maximum continuous</td><td>240</td><td>3200</td><td>42.0</td><td>15000</td></tr><tr><td>Takeoff</td><td>260</td><td>3200</td><td>41.0</td><td>S.L.</td></tr><tr><td>Takeoff</td><td>260</td><td>3200</td><td>43.0</td><td>10000</td></tr></table> |                         | HP                      | RPM             | IN.HG.           | ALT. FT.        | Maximum continuous | 240 | 3200 | 38.0 | S.L. | Maximum continuous | 240 | 3200 | 42.0 | 15000 | Takeoff | 260 | 3200 | 41.0 | S.L. | Takeoff | 260 | 3200 | 43.0 | 10000 |
|                               | HP   | RPM                     | IN.HG.                  | ALT. FT.        |                  |                 |                    |     |      |      |      |                    |     |      |      |       |         |     |      |      |      |         |     |      |      |       |
| Maximum continuous            | 240  | 3200                    | 38.0                    | S.L.            |                  |                 |                    |     |      |      |      |                    |     |      |      |       |         |     |      |      |      |         |     |      |      |       |
| Maximum continuous            | 240  | 3200                    | 42.0                    | 15000           |                  |                 |                    |     |      |      |      |                    |     |      |      |       |         |     |      |      |      |         |     |      |      |       |
| Takeoff                       | 260  | 3200                    | 41.0                    | S.L.            |                  |                 |                    |     |      |      |      |                    |     |      |      |       |         |     |      |      |      |         |     |      |      |       |
| Takeoff                       | 260  | 3200                    | 43.0                    | 10000           |                  |                 |                    |     |      |      |      |                    |     |      |      |       |         |     |      |      |      |         |     |      |      |       |
| Rotor Limits (I.A.S.)         | <table><tr><td>Power off (Rotor Tach.)</td><td>Power On (Engine Tach.)</td></tr><tr><td>Max. 360 r.p.m.</td><td>Max. 3200 r.p.m.</td></tr><tr><td>Min. 290 r.p.m.</td><td>Min. 3000 r.p.m.</td></tr></table>   | Power off (Rotor Tach.) | Power On (Engine Tach.) | Max. 360 r.p.m. | Max. 3200 r.p.m. | Min. 290 r.p.m. | Min. 3000 r.p.m.   |     |      |      |      |                    |     |      |      |       |         |     |      |      |      |         |     |      |      |       |
| Power off (Rotor Tach.)       | Power On (Engine Tach.)  |                         |                         |                 |                  |                 |                    |     |      |      |      |                    |     |      |      |       |         |     |      |      |      |         |     |      |      |       |
| Max. 360 r.p.m.               | Max. 3200 r.p.m.   |                         |                         |                 |                  |                 |                    |     |      |      |      |                    |     |      |      |       |         |     |      |      |      |         |     |      |      |       |
| Min. 290 r.p.m.               | Min. 3000 r.p.m.   |                         |                         |                 |                  |                 |                    |     |      |      |      |                    |     |      |      |       |         |     |      |      |      |         |     |      |      |       |
| Airspeed limits               | Vne (Never-exceed speed) with skids 86 knots (S.L.)<br>Vne (Never-exceed speed) with floats 65 knots (S.L.)<br>For reduction of Vne with altitude, see Rotorcraft Flight Manual  |                         |                         |                 |                  |                 |                    |     |      |      |      |                    |     |      |      |       |         |     |      |      |      |         |     |      |      |       |
| C.G. range<br>(Longitudinal). | 94.5 to 101.6 in.  |                         |                         |                 |                  |                 |                    |     |      |      |      |                    |     |      |      |       |         |     |      |      |      |         |     |      |      |       |
| Empty weight C.G. range       | None   |                         |                         |                 |                  |                 |                    |     |      |      |      |                    |     |      |      |       |         |     |      |      |      |         |     |      |      |       |
| Datum                         | 98.4 in. forward of rotorshaft centerline  |                         |                         |                 |                  |                 |                    |     |      |      |      |                    |     |      |      |       |         |     |      |      |      |         |     |      |      |       |
| Leveling means                | Rotorhead of lower rotor   |                         |                         |                 |                  |                 |                    |     |      |      |      |                    |     |      |      |       |         |     |      |      |      |         |     |      |      |       |
| Maximum weight                | 3300 lb.   |                         |                         |                 |                  |                 |                    |     |      |      |      |                    |     |      |      |       |         |     |      |      |      |         |     |      |      |       |
| No. of seats                  | 1 at (66.9)  |                         |                         |                 |                  |                 |                    |     |      |      |      |                    |     |      |      |       |         |     |      |      |      |         |     |      |      |       |
| Maximum baggage               | None. Cargo accommodations provided in accordance with LBA-approved Service Kit installations listed in Section 7 of LBA-approved Rotorcraft Flight Manual.  |                         |                         |                 |                  |                 |                    |     |      |      |      |                    |     |      |      |       |         |     |      |      |      |         |     |      |      |       |
| Fuel capacity                 | 26.1 gal (total) 1 Tank at (81.5). See NOTE 1 for unusable fuel  |                         |                         |                 |                  |                 |                    |     |      |      |      |                    |     |      |      |       |         |     |      |      |      |         |     |      |      |       |
| Oil capacity                  | 1.4 gal. at (181.3). See NOTE 1 for unusable oil   |                         |                         |                 |                  |                 |                    |     |      |      |      |                    |     |      |      |       |         |     |      |      |      |         |     |      |      |       |
| Other Operating Limits        | See LBA-approved Rotorcraft Flight Manual  |                         |                         |                 |                  |                 |                    |     |      |      |      |                    |     |      |      |       |         |     |      |      |      |         |     |      |      |       |

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| Main Rotor Blade Movements | (Helicopter level. The angles refer to upper surfaces of rotor grips)  |              |        |
|                            | Collective pitch   | +2° to + 10° | (±30') |
|                            | Cyclic forward   | 7°           | (±30') |
|                            | aft  | 7°           | (±30') |
|                            | left   | 4°           | (±30') |
|                            | right  | 4°           | (±30') |
|                            | Differential for Yaw Control   |              |        |
|                            | right  | 2°           | (±10') |
|                            | left   | 2°           | (±10') |
| Serial Nos. Eligible       | The Federal Republic of Germany Certificate of Airworthiness for Export endorsed as noted below under "Import Requirements", must be submitted for each individual Rotorcraft for which application for FAA certification is made.   |              |        |
| Import Requirements        | A U.S. Airworthiness Certificate may be issued on the basis of a Certificate of Airworthiness for Export signed by a representative of the Luftfahrt-Bundesamt (LBA) containing the following statement: "The Rotorcraft covered by this certificate has been examined, tested, and found to conform to the type design approved under Type Certificate No. H5EU, and to be in condition for safe operation".  |              |        |
| Certification Basis        | FAR 27 effective 1 February 1965 including Amendments 27-1 through 27-4.<br>Type Certificate Import No. H5EU issued 19 October 1972.<br>Date of application for Type Certificate: 9 May 1969.  |              |        |
| Equipment:                 | The basic required equipment as prescribed in the applicable Airworthiness Regulations (See Certification basis) must be installed in the Rotorcraft for certification. In addition, the following is required:<br>(a) Current LBA approved FJ Sky-Trac Rotorcraft Flight Manual<br>(Reissued 30 May 1972).  |              |        |
| NOTE 1.                    | Current weight and balance report including list of equipment included in certificated empty weight, and loading instructions, must be provided for each helicopter at the time of original airworthiness certification. Ballast, when necessary, must be carried in accordance with the loading instructions of the Rotorcraft Flight Manual.<br><br>The certificated empty weight and corresponding Center of Gravity location must include:<br><br>Unusable fuel 7.7 lbs. at (81.5)<br>Unusable oil 5.1 lbs. at (181.3) |              |        |
| NOTE 2.                    | The following placard must be installed in front and clear view of the pilot:<br><br>"This Helicopter must be operated in compliance with the operating limitations specified in the LBA-approved Rotorcraft Flight Manual. The "Airworthiness Limitations" section of the Rotorcraft Maintenance Manual must be complied with".<br><br>All placards listed in the LBA-approved Rotorcraft Flight Manual must be installed in the appropriate locations.   |              |        |
| NOTE 3.                    | Replacement times of helicopter structural parts which are fatigue critical are listed in the "Airworthiness Limitations" section of the Maintenance Manual.   |              |        |
| NOTE 4.                    | This helicopter must be serviced and maintained in compliance with instructions given by Helicopter Technik Munchen in the inspection Requirements Manual and in the Maintenance Manual.   |              |        |

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