

**DEPARTMENT OF TRANSPORTATION
FEDERAL AVIATION ADMINISTRATION**

H80EU
Revision 2
AgustaWestland
EH101-300
February 28, 2022

TYPE CERTIFICATE DATA SHEET NO. H80EU

This data sheet, which is part of Type Certificate No. H80EU prescribes conditions and limitations under which the product for which the Type Certificate was issued meets the airworthiness requirements of the Federal Aviation Regulations.

Type Certificate Holder. AGUSTAWESTLAND

Company controlled jointly by:

AGUSTA ELI s.r.l.
21017 - Cascina Costa di Samarate (Varese) Italia
and
WESTLAND HELICOPTER LIMITED
Yeovil, Somerset, England

I. Model EH101-300 (Transport Category Helicopter) approved November 25, 1994.

Engines. Three General Electric model CT7-6 turboshaft engines with DECU.
(Ref. CID 618776 and CID 618775).
One auxiliary power unit Sunstrand model T-62T40C7EH/
P/N 4502316 with ESU P/N 4502145.

Fuel. Jet A - ASTM D1655
Jet A1 - ASTM D1655
Jet B - ASTM D1655

Fuel system icing inhibitor is to be used below + 5°C OAT. See U.K.
CAA approved Rotorcraft Flight Manual.

Powerplant Limitations.

All engine operation:

- Maximum Continuous

Torque (transmission rating)	100% (1254 lb-ft)	(4884 shp at 100% rpm; 5006 shp at 102.5% rpm)
Output shaft speed (Nf)	102.5% (20974 rpm)	
Gas Producer Speed (Ng)	101.6% (45415 rpm)	
Gas Temperature	899 °C (1650 °F)	

- Take Off (5 minutes)

Torque (Transmission rating)	106.5% (1335 lb-ft)	(5199 shp at 100% rpm; 5329 shp at 102.5% rpm)
Output shaft speed (Nf)	102.5% (20974 rpm)	
Gas Producer Speed (Ng)	102.6% (45862 rpm)	

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Gas Temperature 948 °C (1738 °F)
 One Engine Inoperative (Emergency)

- Maximum Continuous

Torque (transmission rating)	112 % (934 lb-ft)	(3640 shp at 100% rpm; 3731 shp at 102.5% rpm)
Output shaft speed (Nf)	102.5% (20974 rpm)	
Gas Producer Speed (Ng)	102.6% (45862 rpm)	
Gas Temperature	948 °C (1738 °F)	

- 2.5 Minutes Rating

Torque (transmission rating)	118% (986 lb-ft)	(3840 shp at 100% rpm; 3936 shp at 102.5% rpm)
Output shaft speed (Nf)	102.5% (20974 rpm)	
Gas Producer Speed (Ng)	103% (46041 rpm)	
Gas Temperature	964 °C (1767 °F)	

Rotor Limits.

Power off
 Maximum 110% (231 rpm)
 Minimum 95% (199.5 rpm)

Power on-Take Off and Landing
 Maximum 103% (216.3 rpm)
 Minimum 98% (205.8 rpm)

Power on-Cruise
 Maximum 101% (212.1 rpm)
 Minimum 98% (205.8 rpm)

- Rotor Speed Warning

Low speed p. on	96%	(201.6 rpm)
High speed p. on	105%	(220.5 rpm)
Low speed p. off	95%	(199.5 rpm)
High speed p. off	110%	(231 rpm)

Airspeed Limits.

Never exceed speed (Vne) 167 Kts IAS

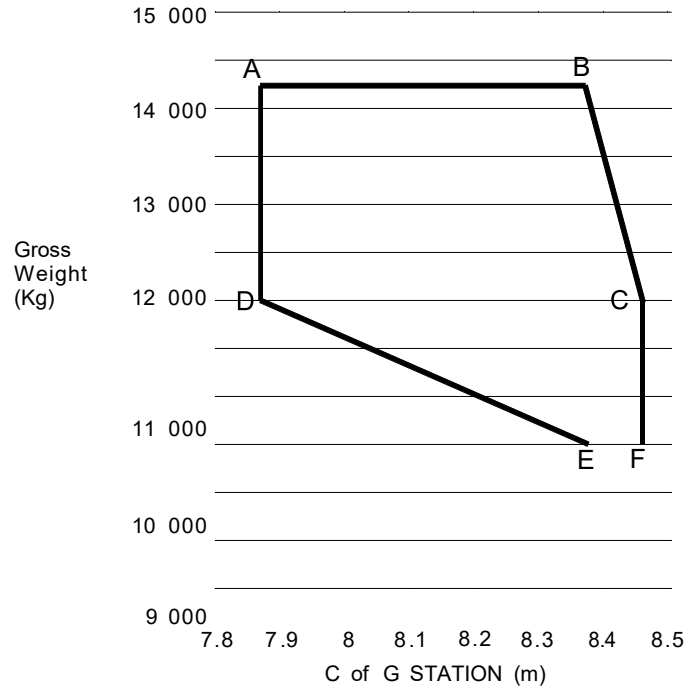
For reduction of Vne with altitude
 OAT and weight: see U.K. CAA
 approved Rotorcraft Flight Manual.

Maximum Gear Operating Speed (Vlo)	167 Kts IAS
Maximum Gear Extended Speed (Vle)	167 kts IAS

Never Exceed Speed Power Off (Vne power off)	134 Kts IAS
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Center of Gravity (CG) Range.

Longitudinal limits



kg	m	kg	m
A 14290	- 7.875	D 12000	- 7.875
B 14290	- 8.385	E 10500	- 8.385
C 12000	- 8.460	F 10500	- 8.460

Lateral center of gravity limits are ± 30 mm from the centerline at all gross weights.

Empty Weight CG Range.

From Sta 8385 mm (330.12 in) to Sta 8460 mm (333.07 in.)

Datum.

Longitudinal station 0 (datum) is 3385 mm (133.26 in) forward of the front jack point.

Lateral station 0 (datum) is ± 1400 mm (55.12 in) inboard of each main jack point and coincides with the rotorcraft longitudinal plane of symmetry.

Leveling Means.

Plumb line from ceiling reference point to index plate on floor of passenger cabin.

Maximum Weight.

14290 kg (31504 lb)

Minimum Crew.

Two (2) pilots.

Number of Passenger Seats.

Thirty (30) passengers.

Maximum Baggage.

Not applicable.

<u>Fuel Capacity.</u>	<p>Total: 1118.88 U.S. Gal. (4235.42 Liters) in 4 tanks of 279.72 U.S. Gal.(1058.86 Liters) each at Sta. 211.6; 250.98; 290.35; 329.72 in (5374.64; 6374.892; 7374.89; 8374.888 mm) respectively</p> <p>Usable: 1097.75 U.S. Gal. (4155.44 Liters). See Note 1 for unusable fuel.</p>
<u>Oil Capacity.</u>	<p><u>Engines</u> 1.63 U.S. Gal. (6.17 Liters) each engine, two at Sta. 330.86 in (8403.84 mm) and one at Sta. 381.96 in (9701.784 mm)</p> <p><u>APU</u> 0.79 U.S. Gal. (2.99 Liters) at Sta. 395.4 in (10043.16 mm)</p> <p><u>Main Gearbox</u> 13.2 U.S. Gal. (49.97 Liters) at Sta. 320.47 in (8139.94 mm)</p> <p><u>Accessory Gearbox</u> 1.82 U.S. Gal. (6.89 Liters) at Sta. 266.8 in. (6776.72 mm)</p> <p><u>Intermediate Gearbox</u> 0.76 U.S. Gal. (2.88 Liters) at Sta. 736.6 in (18709.64 mm)</p> <p><u>Tail Gearbox</u> 0.92 U.S. Gal. (3.48 Liters) at Sta. 763.5 in (19392.90 mm). See Note 1 for undrainable oil.</p>
<u>Maximum Operating Altitude.</u>	10000 ft (3048 m)
<u>Rotor Blade and Control Movements.</u>	For rigging information refer to the EH101 Maintenance Manual Document # EC02P002J.
<u>Serial Numbers Eligible.</u>	50008
<u>Import Requirements.</u>	<p>To be considered eligible for operation in the United States, each aircraft manufactured under this Type Certificate must be accompanied by a certificate of airworthiness for export or certifying statement endorsed by the exporting civil airworthiness authority which states (in the English language): "This aircraft conforms to its U.S. type design (Type Certificate Number H80EU) and is in a condition for safe operation. The U.S. airworthiness certification basis for aircraft type certificated under FAR Section 21.29 and exported by the country of manufacture is FAR Sections 21.183(c) or 21.185(c).</p> <p>Refer to the applicable bilateral agreement to verify eligibility for import into the United States of both new and used aircraft based on the scope of the agreement, to identify any required statements by the exporting authority on the export certificate of airworthiness (or equivalent document), and for procedures for coordinating exceptions to conformity statements on these documents. Refer to FAA Order 8130.2, <i>Airworthiness Certification of Aircraft</i>, for requirements for issuance of an <i>airworthiness certificate</i> for imported aircraft.</p> <p>The U.S. airworthiness certification basis for aircraft type certificated under FAR Section 21.29 exported from countries other than the country of manufacture (e.g. third party country) is FAR Sections 21.183(d) or 21.183(b).</p> <p>Certificates of Airworthiness under this Type Certificate may not be granted unless approved passenger seating featuring upper torso restraint is installed.</p> <p><u>Note:</u> The Maintenance Manual for the EH101-300 has not been accepted by the FAA; therefore, this aircraft is not eligible for</p>

issuance of a standard certificate of airworthiness. For further information, contact AEU-100 FAA/EMB, PSC 82 Box 002, APO AE 09724 or FTW-AEG, DOT/FAA, Fort Worth, TX 76193-0270

Certification Basis.

FAR 21 including Amendment 21-61.

FAR 29 including Amendment 29-1 through 29-27.

FAR 29.351 Amendment 29-30.

Equivalent Safety Findings for FAR 29.903(b) and FAR 29.1303(g)(2)

Special condition no. 29-ASW-12 on HIRF.

FAR 36 including Amendment 36-14.

Civil Aviation Authority-United Kingdom (CAA-U.K.) originally type certificated this rotorcraft under its type certificate number BR2. The FAA validated this product under U.S. Type Certificate Number H80EU

Equipment.

The basic required equipment as prescribed in the applicable airworthiness regulations (see certification basis) must be installed in the aircraft for certification.

In addition a CAA approved Rotorcraft Flight Manual is required.

NOTES.

The FAA can issue a U.S. airworthiness certificate based on an National Aviation Authority (NAA) Export Certificate of Airworthiness (Export C of A) signed by a representative of the Civil Aviation Authority-United Kingdom (CAA-UK) on behalf of the European Community.

The Export C of A should contain the following statement: "The aircraft covered by this certificate has been examined, tested, and found to comply with the type design" approved under U.S. Type Certificate Number H80EU and to be in a condition for safe operation.

Note 1.

Current weight and balance report including list of equipment included in certificated empty weight, and loading instructions when necessary must be provided for each aircraft at the time of original certification.

The certificated empty weight and corresponding CG location must include total oil capacity of 22.43 U.S. Gal. (84.9 lt) at sta. 9074 mm (357.24 in) and unusable fuel of 21.03 U.S. Gal. (79.6 lt) at sta. 7259 mm (285.8 in).

Note 2.

Placards listed in the CAA approved Rotorcraft Flight Manual must be displayed in the appropriate location.

Note 3.

Instructions for continued airworthiness of the rotorcraft are provided in the following documentation:

- Document No. ED02P21IJTC Continued Airworthiness Schedule and Airworthiness Limitations Manual

- Document No. EC02P002J Maintenance Manual

which must be supplied with each rotorcraft at time of delivery. Service bulletins, structural repair manuals, overhaul and maintenance manuals, which contain a statement that the document is CAA approved, are accepted by the FAA and are considered FAA approved. These acceptances pertain to the type design only.

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