

DEPARTMENT OF TRANSPORTATION
FEDERAL AVIATION ADMINISTRATION

	A7CE
	Revision 50
	Textron Aviation Inc.
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	February 21, 2020

TYPE CERTIFICATE DATA SHEET NO. A7CE

This data sheet which is part of Type Certificate No. A7CE prescribes conditions and limitations under which the product for which the type certificate was issued meets the airworthiness requirements of the Federal Aviation Regulations.

Type Certificate Holder Textron Aviation Inc.
One Cessna Boulevard
Wichita, Kansas 67215

Type Certificate Holder Record Cessna Aircraft Company transferred to
Textron Aviation Inc. on July 29, 2015

I. Model 411 (Normal Category), Approved August 17, 1964
Model 411A (Normal Category), Approved January 26, 1967

Engines Two Continental GTSIO-520-C, reduction gear ratio .750:1

Fuel Grade 100 or 100LL aviation gasoline

Engine Limits For all operations, 2400 propeller r.p.m. (340 hp.)
34.5 in. Hg. Mp. up to critical altitude of 16,000 ft. in standard atmosphere. Above
16,000 ft. the following maximum Mp. applies for maximum r.p.m.

<u>Altitude (ft.)</u>	<u>Max. Allowable Mp. (in. Hg.)</u>
16,000	34.5
18,000	31.2
20,000	29.0
22,000	26.4
24,000	24.3
26,000	22.2
28,000	20.2
30,000	18.5

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I. Models 411, 411A (cont'd)Propeller and
Propeller Limits

1. Model 411 only
Two Hartzell full-feathering 3-bladed propeller installations
(a) Hartzell Hub HC-A3VF-2D with V8833 blades
Diameter: not over 88.4 in., not under 86.4 in.
(no further reduction permitted)
Pitch settings at 30 in. station:
low 14.0°, +0°, -2°
feathered 84.0°, +2°, -0°
(b) Hydraulic Governor Woodward A210444, 210439, C210446 or B210529
(c) Propeller spinner and bulkhead assembly, Hartzell 835-20
- or 2. Models 411, 411A
Two McCauley full-feathered 3-bladed propeller installations
(a) McCauley hub 3AF34C74 with 90LF-0 blades or
McCauley hub 3AF37C510 with 90LFB blades
Diameter: not over 90 in., not under 84.0 in. with
90LF-0 blades or not under 88.0 in. with 90LFB-0 blades.
(no further reduction permitted)
Pitch settings at 30 in. station:
low 14.0°, ±0.2°
feathering 84.5°, ±0.3°
(b) Hydraulic governor Woodward A210444, 210439, C210446 or B210529
(c) Propeller spinner and bulkhead assembly,
McCauley D-3574 or D-3732 for use with C74 Model Propeller, or
McCauley D-7229 for use with C510 Model Propeller.

Airspeed Limits
(CAS)

Maneuvering	180 m.p.h. (156 knots)
Maximum structural cruising	230 m.p.h. (200 knots)
Never exceed	266 m.p.h. (231 knots)
Landing gear operating	160 m.p.h. (139 knots)
Landing gear extended	160 m.p.h. (139 knots)
Flaps extended 15°	180 m.p.h. (156 knots)
Flaps extended 45°	160 m.p.h. (139 knots)
Minimum control	103 m.p.h. (90 knots)

C.G. Range (Landing
Gear Extended)

(+150.6) to (+155.5) at 6500 lb.
(+155.7) at 6100 lb. or less
(+144.3) at 5200 lb. or less
Straight line variation between points given
Landing gear retracted moment change: +837 in.-lb.

Empty Wt. C.G. Range

None

Leveling Means

External screw heads on right side of fuselage at stations +213.65 and +238.00
on W.L. +93.80

Maximum Weight

Landing 6500 lb., takeoff 6500 lb.

No. of Seats

6, 7 or 8 (2 at +137.0, 2 at +175.5, 2 at +215.5, 1 or 2 at +238.0)
(See manufacturer's equipment list for optional seating arrangements)

Maximum Baggage

Model 411: 120 lb. (+58.0), 240 lb. (+186.0), 340 lb. (+246.5)
Model 411A: 350 lb. (+71.0), 240 lb. (+186.0), 340 lb. (+246.5)

Fuel Capacity

175 gal. (2 wing tip tanks, 51 gal. ea., 50 gal. usable at
+152.0 and 2 wing tanks, 36.5 gal. ea., 35 gal. usable at +164.0)
See NOTE 1 for data on unusable fuel

Oil Capacity

26 qt. (13 qt. in ea. engine at +115.4; usable 7.0 qt. per engine)
See NOTE 1 for undrainable oil

I. Models 411, 411A (cont'd)

Control Surface Movements	Wing flaps			Down	45°, +1°, -0°
	Main surfaces				
	Aileron	Up	20°, +1°, -0°	Down	20°, +1°, -0°
	Elevator	Up	25°, +1°, -0°	Down	15°, +1°, -0°
	Rudder	Right	32°, +1°, -0°	Left	32°, +1°, -0°
	(Read degrees normal to rudder hinge line)				
	Tab (main surface in neutral)				
	Aileron	Up	20°, +1°, -0°	Down	20°, +1°, -0°
	Elevator	Up	10°, +1°, -0°	Down	26°, +1°, -0°
	Rudder	Right	17°, +1°, -0°	Left	22°, +1°, -0°
(Read degrees normal to rudder hinge line)					
Serial Nos. Eligible	Model 411:	411-0001 through 411-0250			
	Model 411A:	411-0251 through 411-0300			

II. Model 401 (Normal Category), Approved September 20, 1966**Model 401A (Normal Category), Approved October 29, 1968****Model 401B (Normal Category), Approved November 12, 1969**

Engines	Two Continental TSIO-520-E or TSIO-520-EB (In any combination)
Fuel	Grade 100 or 100LL aviation gasoline
Engine Limits	For all operations, 2700 r.p.m. (300 hp.) 34.5 in. Hg. Mp. up to critical altitude of 16,000 ft. in standard atmosphere. Above 16,000 ft. the following maximum Mp. applies for maximum r.p.m.

<u>Altitude (ft.)</u>	<u>Max Allowable Mp. (in. Hg.)</u>
16,000	34.5
18,000	31.8
20,000	29.5
22,000	27.3
24,000	25.1
26,000	23.0
28,000	22.0
30,000	19.0

Propeller and Propeller Limits	Two McCauley full-feathered 3-bladed propeller installations
	(a) McCauley hub 3AF32C87 with 82NC-5.5 blades or McCauley hub 3AF32C504 with 82NEA-5.5 blades
	Diameter: not over 76.5 in., not under 74.0 in. (no further reduction permitted)
	Pitch settings at 30 in. station: low 14.2°, ±0.2° feathered 81.2°, ±0.3°
	(b) <u>Model 401:</u> Hydraulic Governor Woodward B210444, C210439, B210446 or A210529F <u>Model 401A, 401B:</u> Hydraulic Governor Woodward B210444, C210439, B210446, or A210529F McCauley DCF290D1/T3, DCF290D2/T3, DCF290D7/T3, DCFU290D1/T3, DCFU290D2/T3, DCFU290D7/T3, DCFU290D13/T3, DCFS290D1/T3, DCFS290D2/T3, DCFS290D7/T3, DCFUS290D1/T3, DCFUS290D2/T3, DCFUS290D7/T3, DCFUS290D13/T3.
	(c) Propeller spinner and bulkhead assembly, McCauley D-3534/D-3537, D-3534/D-3796, and D-5212/D5214.

II. Models 401, 401A, 401B (cont'd)

Airspeed Limits (CAS)	Maneuvering	180 m.p.h. (156 knots)
	Maximum structural cruising	230 m.p.h. (200 knots)
	Never exceed	266 m.p.h. (231 knots)
	Landing gear operating	160 m.p.h. (139 knots)
	Landing gear extended	160 m.p.h. (139 knots)
	Flaps extended 15°	180 m.p.h. (156 knots)
	Flaps extended 45°	160 m.p.h. (139 knots)
	Minimum control	95 m.p.h. (83 knots)
C.G. Range (Landing Gear Extended)	(+150.8) to (+158.1) at 6300 lb. (+158.5) at 5900 lb. or less (+147.5) at 5000 lb. or less Straight line variation between points given Landing gear retracted moment change: +837 in.-lb.	
Empty Wt. C.G. Range	None	
Leveling Means	External screw heads on right side of fuselage at stations +213.65 and +238.00 on W.L. +93.80	
Maximum Weight	Landing 6200 lb., takeoff 6300 lb.	
No. of Seats	6, 7 or 8 (2 at +137.0, 2 at +175.6, 2 at +215.5, 1 or 2 at +238.0) (See manufacturer's equipment list for optional seating arrangements)	
Maximum Baggage	350 lb. (+71.0), 240 lb. (+186.0), 340 lb. (+246.5)	
Fuel Capacity	102 gal. (2 wing tip tanks, 51 gal. ea., 50 gal. usable at +152.0) See NOTE 1 for data on unusable fuel	
Oil Capacity	26 qt. (13 qt. in ea. engine at +113.5; usable 6.5 qt. per engine) See NOTE 1 for data on undrainable oil	
Control Surface Movements	Wing flaps	Down 45°, +1°, -0°
	Main surfaces	
	Aileron Up 20°, +1°, -0°	Down 20°, +1°, -0°
	Elevator Up 25°, +1°, -0°	Down 15°, +1°, -0°
	Rudder Right 32°, +1°, -0°	Left 32°, +1°, -0°
	(Read degrees normal to rudder hinge line)	
	Tab (main surface in neutral)	
	Aileron Up 20°, +1°, -0°	Down 20°, +1°, -0°
	Elevator Up 5°, +1°, -0°	Down 30°, +1°, -0°
	Rudder Right 7°, +1°, -0°	Left 9°, +1°, -0°
	(Read degrees normal to rudder hinge line)	
Serial Nos. Eligible	Model 401: 401-0001 through 401-0322	
	Model 401A: 401A0001 through 401A0132	
	Model 401B: 401B0001 through 401B0221	

III. Model 402 (Normal Category), Approved September 20, 1966
Model 402A (Normal Category), Approved January 3, 1969
Model 402B (Normal Category), Approved November 12, 1969

Engines	Two Continental TSIO-520-E or TSIO-520-EB (in any combination)
Fuel	Grade 100 or 100LL aviation gasoline

III. Models 402, 402A, 402B (cont'd)

Engine Limits

For all operations, 2700 r.p.m. (300 hp.)
 34.5 in. Hg. Mp. up to critical altitude of 16,000 ft. in standard atmosphere. Above
 16,000 ft. the following maximum Mp. applies for maximum r.p.m.

<u>Altitude (ft.)</u>	<u>Max. Allowable Mp. (in. Hg.)</u>
16,000	34.5
18,000	31.8
20,000	29.5
22,000	27.3
24,000	25.1
26,000	23.0
28,000	22.0
30,000	19.0

Propeller and
Propeller Limits

Two McCauley full-feathered 3-bladed propeller installations

(a) McCauley hub 3AF32C87 with 82NC-5.5 blades or McCauley hub
 3AF32C504 with 82NEA-5.5 blades

Diameter: not over 76.5 in., not under 74.0 in.

(no further reduction permitted)

Pitch settings at 30 in. station:

low 14.2°, $\pm 0.2^\circ$

feathering 81.2°, $\pm 0.3^\circ$

(b) Model 402, 402A, 402B, S/N 402B0001 thru 402B1200

Hydraulic governor, Woodward B210444, C210439, B210446F or
 A210529H; McCauley DCF290D1/T3, DCF290D2/T3, DCFS290D1/T3,
 DCFS290D2/T3, DCFU290D1/T3, DCFU290D2/T3, DCFUS290D1/T3,
 DCFUS290D2/T3, DCF290D7/T3, DCFS290D7/T3, DCFU290D7/T3,
 DCFU290D13/T3, DCFUS290D7/T3 or DCFUS290D13/T3.

Model 402B, S/N 402B1201 through 402B1300

Hydraulic governor, Woodward B210444, C210439; McCauley
 DCF290D1/T3, DCF290D2/T3, DCFU290D1/T3, DCFU290D2/T3,
 DCFS290D4/T3, DCFUS290D4/T3, DCFS290D5/T3, DCFUS290D5/T3,
 DCF290D7/T3, DCFU290D7/T3, DCFS290D7/T3, DCFUS290D7/T3,
 DCFU290D13/T3 or DCFUS290D13/T3. **For aircraft modified by SK414-10B,
 DCFS290D6/T3, DCFUS290D6/T3, DCFS290D8/T3 or DCFUS290D8/T3.**

Model 402B, S/N 402B1301 and On

Hydraulic governor, Woodward B210444, C210439; McCauley
 DCF290D1/T3, DCF290D2/T3, DCFU290D1/T3, DCFU290D2/T3,
 DCFS290D4/T3, DCFUS290D4/T3, DCFS290D6/T3, DCFUS290D6/T3,
 DCF290D7/T3, DCFU290D7/T3, DCFS290D7/T3, DCFUS290D7/T3,
 DCFS290D8/T3, DCFUS290D8/T3, DCFU290D13/T3, DCFUS290D12/T3
 or DCFUS290D13/T3.

(c) Propeller spinner and bulkhead assembly, McCauley D-3534/D-3537,
 D-3534/D-3796 or D-5212/D5214.

III. Models 402, 402A, 402B (cont'd)

Airspeed Limits (CAS)	Model 402, S/N 402-0001 and On
	Model 402A, S/N 402A0001 and On
	<u>Model 402B, S/N 402B0001 through 402B0500</u>
	Maneuvering 180 m.p.h. (156 knots)
	Maximum structural cruising 230 m.p.h. (200 knots)
	Never exceed 266 m.p.h. (231 knots)
	Landing gear operating 160 m.p.h. (139 knots)
	Landing gear extended 160 m.p.h. (139 knots)
	Flaps extended 15° 180 m.p.h. (156 knots)
	Flaps extended 45° 160 m.p.h. (139 knots)
	Minimum control 95 m.p.h. (83 knots)
	<u>Model 402B, S/N 402B0501 through 402B1000</u>
	Maneuvering 156 KCAS (180 m.p.h.)
	Maximum structural cruising 200 KCAS (230 m.p.h.)
	Never exceed 231 KCAS (266 m.p.h.)
	Landing gear operating 140 KCAS (161 m.p.h.)
	Landing gear extended 140 KCAS (161 m.p.h.)
	Flaps extended 15° 160 KCAS (184 m.p.h.)
	Flaps extended 45° 140 KCAS (161 m.p.h.)
	Minimum control 83 KCAS (95 m.p.h.)
(IAS)	<u>Model 402B, S/N 402B1001 and On</u>
	Maneuvering 156 KIAS (180 m.p.h.)
	Maximum structural cruising 199 KIAS (229 m.p.h.)
	Never exceed 230 KIAS (265 m.p.h.)
	Landing gear operating 140 KIAS (161 m.p.h.)
	Landing gear extended 140 KIAS (161 m.p.h.)
	Flaps extended 15° 160 KIAS (184 m.p.h.)
	Flaps extended 45° 140 KIAS (161 m.p.h.)
	Minimum control 82 KIAS (94 m.p.h.)
C.G. Range (Landing Gear Extended)	(+150.8) to (+159.7) at 6300 lb.
	(+160.2) at 5900 lb. or less
	(+147.5) at 5000 lb. or less
	Straight line variation between points given
	Landing gear retracted moment change: +837 in.-lb.
Empty Wt. C.G. Range	None
Leveling Means	External screw heads on right side of fuselage at stations +213.65 and +238.00 on W.L. +93.80
Maximum Weight	<u>Models 402, 402A, 402B, S/N 402B0001 through 402B1300</u>
	Landing 6200 lb., takeoff 6300 lb.
	<u>Model 402B, S/N 402B1301 and On</u>
	Landing 6200 lb., ramp 6335 lb., takeoff 6300 lb.
No. of Seats	<u>Model 402</u>
	9 (2 at +137.0, 2 at +166.0, 2 at +193.0, 2 at +220.0, 1 at +247.0)
	<u>Model 402A, 402B, S/N 402B0001 through 402B0300</u>
	9 or 10 (2 at +137.0, 2 at +166.0, 2 at +193.0, 2 at +220.0, 1 or 2 at +247.0)
	<u>Model 402B, S/N 402B0301 and On</u>
	6, 7 or 8 (2 at +137.0, 2 at +175.0, 2 at +218.0, 1 or 2 at +261.0)
	9 (with photographic provisions option) (2 at +137.0, 2 at +162.0, 2 at +190.0, 2 at +218.0, 1 at +246.0)
	10 (2 at +137.0, 2 at +162.0, 2 at +190.0, 2 at +218.0, 2 at +246.0)
	(See manufacturer's equipment list for optional seating arrangements)

III. Models 402, 402A, 402B (cont'd)

Maximum Baggage	<u>Models 402, 402A, 402B, S/N 402B0001 through 402B0300</u> 350 lb. (+71.0), 240 lb. (+186.0), 170 lb. (+247.0)																																																										
	<u>Model 402B, S/N 402B0301 and On</u> 250 lb. (+32.0), 350 lb. (+71.0), 240 lb. (+186.0), 400 lb. (+266.0), 100 lb. (+282.0)																																																										
Fuel Capacity	102 gal. (2 wing tip tanks, 51 gal. ea., 50 gal. usable at +152.0) See NOTE 1 for data on unusable fuel																																																										
Oil Capacity	26 qt. (13 qt. in ea. engine at +113.5; usable 6.5 qt. per engine) See NOTE 1 for data on undrainable oil																																																										
Control Surface Movements	<table> <tr> <td>Wing flaps</td><td></td><td></td><td>Down</td><td>45°, +1°, -0°</td></tr> <tr> <td>Main surfaces</td><td></td><td></td><td></td><td></td></tr> <tr> <td>Aileron</td><td>Up</td><td>20°, +1°, -0°</td><td>Down</td><td>20°, +1°, -0°</td></tr> <tr> <td>Elevator</td><td>Up</td><td>25°, +1°, -0°</td><td>Down</td><td>15°, +1°, -0°</td></tr> <tr> <td>Rudder</td><td>Right</td><td>32°, +1°, -0°</td><td>Left</td><td>32°, +1°, -0°</td></tr> <tr> <td colspan="5">(Read degrees normal to rudder hinge line)</td></tr> <tr> <td>Tab (main surface in neutral)</td><td></td><td></td><td></td><td></td></tr> <tr> <td>Aileron</td><td>Up</td><td>20°, +1°, -0°</td><td>Down</td><td>20°, +1°, -0°</td></tr> <tr> <td>Elevator</td><td>Up</td><td>5°, +1°, -0°</td><td>Down</td><td>30°, +1°, -0°</td></tr> <tr> <td>Rudder</td><td>Right</td><td>7°, +1°, -0°</td><td>Left</td><td>9°, +1°, -0°</td></tr> <tr> <td colspan="5">(Read degrees normal to rudder hinge line)</td></tr> </table>				Wing flaps			Down	45°, +1°, -0°	Main surfaces					Aileron	Up	20°, +1°, -0°	Down	20°, +1°, -0°	Elevator	Up	25°, +1°, -0°	Down	15°, +1°, -0°	Rudder	Right	32°, +1°, -0°	Left	32°, +1°, -0°	(Read degrees normal to rudder hinge line)					Tab (main surface in neutral)					Aileron	Up	20°, +1°, -0°	Down	20°, +1°, -0°	Elevator	Up	5°, +1°, -0°	Down	30°, +1°, -0°	Rudder	Right	7°, +1°, -0°	Left	9°, +1°, -0°	(Read degrees normal to rudder hinge line)				
Wing flaps			Down	45°, +1°, -0°																																																							
Main surfaces																																																											
Aileron	Up	20°, +1°, -0°	Down	20°, +1°, -0°																																																							
Elevator	Up	25°, +1°, -0°	Down	15°, +1°, -0°																																																							
Rudder	Right	32°, +1°, -0°	Left	32°, +1°, -0°																																																							
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Aileron	Up	20°, +1°, -0°	Down	20°, +1°, -0°																																																							
Elevator	Up	5°, +1°, -0°	Down	30°, +1°, -0°																																																							
Rudder	Right	7°, +1°, -0°	Left	9°, +1°, -0°																																																							
(Read degrees normal to rudder hinge line)																																																											
Serial Nos. Eligible	Model 402:	402-0001 through 402-0322																																																									
	Model 402A:	402A0001 through 402A0129																																																									
	Model 402B:	402B0001 through 402B1384																																																									

IV. Model 421 (Normal Category), Approved May 1, 1967
Model 421A (Normal Category), Approved November 19, 1968

Engines	Two Continental GTSIO-520-D, reduction gear ratio .667:1
Fuel	Grade 100 or 100LL aviation gasoline
Engine Limits	For all operations, 2275 propeller r.p.m. (375 hp.) 39.5 in. Hg. Mp. up to critical altitude of 16,000 ft. in standard atmosphere. Above 16,000 ft. the following maximum Mp. applies for maximum r.p.m.

<u>Model 421</u>		<u>Model 421A</u>	
Altitude (ft.)	Max. Allowable Mp. (in. Hg.)	Altitude (ft.)	Max. Allowable Mp. (in. Hg.)
16,000	39.5	16,000	39.5
18,000	32.5	18,000	37.5
20,000	32.5	20,000	35.5
22,000	30.0	22,500	32.5
24,000	27.0	24,000	30.5
26,000	24.5	26,000	28.0
28,000	22.0	28,000	25.5
30,000	20.0	30,000	23.0

IV. Models 421, 421A (cont'd)

Propeller and Propeller Limits	Two McCauley full-feathered 3-bladed propeller installations (a) McCauley hub 3AF34C92 with 90LF-0 blades or McCauley hub 3AF37C516 with 90LFB-0 blades. Diameter: not over 90.0 in., not under 88.0 in. (no further reduction permitted) Pitch settings at 30 in. station: low 16.9°, $\pm 0.2^\circ$ feathering 84.5°, $\pm 0.3^\circ$, (b) Hydraulic Governor Woodward 210594, 210595, 210596, or 210597. (c) Propeller spinner and bulkhead assembly, McCauley D-3573/D-3576, for use with C92 Model propeller, or McCauley D-7229 spinner and bulkhead assembly for use with C516 Model propeller.	
Airspeed Limits (CAS)	Maneuvering	184 m.p.h. (160 knots)
	Maximum structural cruising	230 m.p.h. (200 knots)
	Never exceed	272 m.p.h. (236 knots)
	Landing gear operating	165 m.p.h. (143 knots)
	Landing gear extended	165 m.p.h. (143 knots)
	Flaps extended 15°	180 m.p.h. (156 knots)
	Flaps extended 45°	165 m.p.h. (143 knots)
	Minimum control	106.5 m.p.h. (93 knots)
C.G. Range (Landing Gear Extended)	<u>Model 421</u>	<u>Model 421A</u>
	(+151.9) to (+155.5) at 6800 lb.	(+152.1) to (+155.5) at 6840 lb.
	(+155.7) at 6400 lb. or less	(+155.7) at 6500 lb. or less
	(+144.3) at 5200 lb. or less	(+144.3) at 5200 lb. or less
	Straight line variation between points given Landing gear retracted moment change: +889 in.-lb.	
Empty Wt. C.G. Range	None	
Leveling Means	External screw heads on right side of fuselage at stations +213.29 and +238.55 on W.L. +93.80	
Maximum Weight	<u>Model 421</u> Landing 6500 lb., takeoff 6800 lb. (See NOTE 4 for takeoff 6840 lb.) <u>Model 421A</u> Landing 6500 lb., takeoff 6840 lb.	
No. of Seats	<u>Model 421</u> 6 (2 at +137.0, 2 at +175.5, 2 at +215.5) <u>Model 421A</u> 6 or 7 (2 at +137.0, 2 at +175.5, 2 at +215.5, 1 at +246.5) (See manufacturer's equipment list for optional seating arrangement)	
Maximum Baggage	350 lb. (+71.0), 240 lb. (+186.0), 340 lb. (+246.5)	
Fuel Capacity	175 gal. (2 wing tip tanks, 51 gal. ea., 50 gal. usable at +152.0 and 2 wing tanks, 36.5 gal. ea., 35 gal. usable at +164.0) See NOTE 1 for data on unusable fuel	
Oil Capacity	26 qt. (13 qt. in ea. engine at +115.4; usable 7.0 qt. per engine) See NOTE 1 for data on undrainable oil	

IV. Models 421, 421A (cont'd)

Control Surface Movements	Wing flaps		Down	45°, +1°, -0°
	Main surfaces			
	Aileron	Up	20°, +1°, -0°	Down 20°, +1°, -0°
	Elevator	Up	25°, +1°, -0°	Down 15°, +1°, -0°
	Rudder	Right	25°, +1°, -0°	Left 25°, +1°, -0°
	(Read degrees normal to rudder hinge line)			
	Tab (main surface in neutral)			
	Aileron	Up	20°, +1°, -0°	Down 20°, +1°, -0°
	Elevator	Up	10°, +1°, -0°	Down 26°, +1°, -0°
	Rudder	Right	11°, +1°, -0°	Left 16°, +1°, -0°
(Read degrees normal to rudder hinge line)				
Serial Nos. Eligible	Model 421:	421-0001 through 421-0200		
	Model 421A:	421A0001 through 421A0158		

V. Model 414 (Normal Category), Approved September 24, 1969

Engines	Two Continental TSIO-520-J or TSIO-520-JB (In any combination) (S/N 414-0001 through 414-0800)														
	Two Continental TSIO-520-N or TSIO-520-NB (In any combination) (S/N 414-0801 and up)														
Fuel	Grade 100 or 100LL aviation gasoline														
Engine Limits	For all operations, 2700 r.p.m. (310 hp.) 36.0 in. Hg. Mp. (S/N 414-0001 through 414-0800) 38.0 in. Hg. Mp. (S/N 414-0801 and up) up to critical altitude of 20,000 ft. in standard atmosphere. Above 20,000 ft. the following maximum Mp. applies for maximum r.p.m.														
	<u>S/N 414-0001 through 414-0800</u>														
	<table><tr><td><u>Altitude (ft.)</u></td><td><u>Max. Allowable Mp. (in. Hg.)</u></td></tr><tr><td>20,000</td><td>36.0</td></tr><tr><td>22,000</td><td>33.6</td></tr><tr><td>24,000</td><td>31.2</td></tr><tr><td>26,000</td><td>28.8</td></tr><tr><td>28,000</td><td>26.4</td></tr><tr><td>30,000</td><td>24.0</td></tr></table>	<u>Altitude (ft.)</u>	<u>Max. Allowable Mp. (in. Hg.)</u>	20,000	36.0	22,000	33.6	24,000	31.2	26,000	28.8	28,000	26.4	30,000	24.0
<u>Altitude (ft.)</u>	<u>Max. Allowable Mp. (in. Hg.)</u>														
20,000	36.0														
22,000	33.6														
24,000	31.2														
26,000	28.8														
28,000	26.4														
30,000	24.0														
	<u>S/N 414-0801 and On</u>														
	<table><tr><td><u>Altitude (ft.)</u></td><td><u>Max. Allowable Mp. (in. Hg.)</u></td></tr><tr><td>20,000</td><td>38.0</td></tr><tr><td>22,000</td><td>35.2</td></tr><tr><td>24,000</td><td>32.3</td></tr><tr><td>26,000</td><td>29.8</td></tr><tr><td>28,000</td><td>27.4</td></tr><tr><td>30,000</td><td>25.0</td></tr></table>	<u>Altitude (ft.)</u>	<u>Max. Allowable Mp. (in. Hg.)</u>	20,000	38.0	22,000	35.2	24,000	32.3	26,000	29.8	28,000	27.4	30,000	25.0
<u>Altitude (ft.)</u>	<u>Max. Allowable Mp. (in. Hg.)</u>														
20,000	38.0														
22,000	35.2														
24,000	32.3														
26,000	29.8														
28,000	27.4														
30,000	25.0														

V. **Model 414** (cont'd)

Propeller and Propeller Limits	Two McCauley full-feathered 3-bladed propeller installations	
	(a) McCauley hub 3AF32C93 with 82NC-5.5 blades or McCauley hub 3AF32C505 with 82NEA-5.5 blades	
	Diameter: not over 76.5 in., not under 74.5 in. (S/N 414-0001 through S/N 414-0800), not under 75.0 in. (S/N 414-0801 and On)	
	(no further reduction permitted)	
	Pitch settings at 30 in. station:	
	low 14.9°, $\pm 0.2^\circ$, feathering 81.2°, $\pm 0.3^\circ$	
	(b) <u>Model 414 S/N 414-0001 thru 414-0800</u>	
	Hydraulic governor, Woodward B210444, C210439, B210446F, or A210529H	
	McCauley DCF290D1/T3, DCF290D2/T3, DCF290D7/T3, DCFU290D1/T3, DCFS290D1/T3, DCFUS290D1/T3, DCFS290D2/T3, DCFU290D2/T3, DCFU290D7/T3, DCFU290D13/T3, DCFS290D7/T3, DCFUS290D2/T3, DCFUS290D7/T3 or DCFUS290D13/T3	
	<u>Model 414 S/N 414-0801 and On</u>	
	McCauley DCFS290D4/T3, DCFUS290D4/T3, DCFS290D5/T3, DCFUS290D5/T3, DCFS290D7/T3, or DCFUS290D7/T3, DCFS290D8/T3, DCFUS290D8/T3, DCFUS290D12/T3, or DCFUS290D13/T3	
	(c) Propeller spinner and bulkhead assembly, McCauley D-3534/D-3537, D-3534/D-3796, or D-5212/D-5214.	
Airspeed Limits (CAS)	<u>S/N 414-0001 through 414-0450</u>	
	Maneuvering	180 m.p.h. (156 knots)
	Maximum structural cruising	230 m.p.h. (200 knots)
	Never exceed	266 m.p.h. (231 knots)
	Flaps extended 15°	180 m.p.h. (157 knots)
	Flaps extended 45°	160 m.p.h. (139 knots)
	Landing gear operating	160 m.p.h. (139 knots)
	Landing gear extended	160 m.p.h. (139 knots)
	Minimum control	97 m.p.h. (84 knots)
	<u>S/N 414-0451 through 414-0800</u>	
	Maneuvering	156 KCAS (180 m.p.h.)
	Maximum structural cruising	200 KCAS (230 m.p.h.)
	Never exceed	231 KCAS (266 m.p.h.)
	Flaps extended 15°	160 KCAS (184 m.p.h.)
	Flaps extended 45°	140 KCAS (161 m.p.h.)
	Landing gear operating	140 KCAS (161 m.p.h.)
	Landing gear extended	140 KCAS (161 m.p.h.)
	Minimum control	84 KCAS (97 m.p.h.)
(IAS)	<u>S/N 414-0801 and On</u>	
	Maneuvering	160 KIAS (184 m.p.h.)
	Maximum structural cruising	205 KIAS (236 m.p.h.)
	Never exceed	236 KIAS (272 m.p.h.)
	Flaps extended 15°	164 KIAS (189 m.p.h.)
	Flaps extended 45°	147 KIAS (169 m.p.h.)
	Landing gear operating	143 KIAS (165 m.p.h.)
	Landing gear extended	143 KIAS (165 m.p.h.)
	Minimum control	82 KIAS (94 m.p.h.)
C.G. Range (Landing Gear Extended)	(+150.9) to (+159.7) at 6350 lb. (+160.2) at 5950 lb. or less (+147.5) at 5000 lb. or less Straight line variation between points given Landing gear retracted moment change: +837 in.-lb.	
Empty Wt. C.G. Range	None	

V. Model 414 (cont'd)

Leveling Means	External screw heads on right side of fuselage at stations +213.29 and +238.55 on W.L. +93.80																																														
Maximum Weight	Landing 6200 lb., takeoff 6350 lb.																																														
No. of Seats	<u>S/N 414-0001 through 414-0350</u> 6 or 7 (2 at +137.0, 2 at +175.5, 2 at +215.5, 1 at +246.5) <u>S/N 414-0351 and On</u> 6 (2 at +137.0, 2 at +175.0, 2 at +218.0) 7 (with toilet option) (2 at +137.0, 2 at +175.0, 2 at +218.0, 1 at +250.0) (See manufacturer's equipment list for optional seating arrangements)																																														
Maximum Baggage	<u>S/N 414-0001 through 414-0350</u> 350 lb. (+71.0), 240 lb. (+186.0), 340 lb. (+246.5) <u>S/N 414-0351 and On</u> 350 lb. (+71.0), 240 lb. (+186.0), 400 lb. (+266.0), 100 lb. (+282.0)																																														
Fuel Capacity	102 gal. (2 wing tip tanks, 51 gal. ea., 50 gal. usable at +152.0) See NOTE 1 for data on unusable fuel																																														
Oil Capacity	26 qt. (13 qt. in ea. engine at +113.5; usable 6.5 qt. per engine) See NOTE 1 for data on undrainable oil																																														
Control Surface Movements	<table> <tr> <td colspan="2">Wing flaps</td><td>Down</td><td>45°, +1°, -0°</td></tr> <tr> <td colspan="4">Main surfaces</td></tr> <tr> <td>Aileron</td><td>Up</td><td>20°, +1°, -0°</td><td>Down 20°, +1°, -0°</td></tr> <tr> <td>Elevator</td><td>Up</td><td>25°, +1°, -0°</td><td>Down 15°, +1°, -0°</td></tr> <tr> <td>Rudder</td><td>Right</td><td>32°, +1°, -0°</td><td>Left 32°, +1°, -0°</td></tr> <tr> <td colspan="4">(Read degrees normal to rudder hinge line)</td></tr> <tr> <td colspan="4">Tab (main surface in neutral)</td></tr> <tr> <td>Aileron</td><td>Up</td><td>20°, +1°, -0°</td><td>Down 20°, +1°, -0°</td></tr> <tr> <td>Elevator</td><td>Up</td><td>5°, +1°, -0°</td><td>Down 30°, +1°, -0°</td></tr> <tr> <td>Rudder</td><td>Right</td><td>11°, +1°, -0°</td><td>Left 16°, +1°, -0°</td></tr> <tr> <td colspan="4">(Read degrees normal to rudder hinge line)</td></tr> </table>			Wing flaps		Down	45°, +1°, -0°	Main surfaces				Aileron	Up	20°, +1°, -0°	Down 20°, +1°, -0°	Elevator	Up	25°, +1°, -0°	Down 15°, +1°, -0°	Rudder	Right	32°, +1°, -0°	Left 32°, +1°, -0°	(Read degrees normal to rudder hinge line)				Tab (main surface in neutral)				Aileron	Up	20°, +1°, -0°	Down 20°, +1°, -0°	Elevator	Up	5°, +1°, -0°	Down 30°, +1°, -0°	Rudder	Right	11°, +1°, -0°	Left 16°, +1°, -0°	(Read degrees normal to rudder hinge line)			
Wing flaps		Down	45°, +1°, -0°																																												
Main surfaces																																															
Aileron	Up	20°, +1°, -0°	Down 20°, +1°, -0°																																												
Elevator	Up	25°, +1°, -0°	Down 15°, +1°, -0°																																												
Rudder	Right	32°, +1°, -0°	Left 32°, +1°, -0°																																												
(Read degrees normal to rudder hinge line)																																															
Tab (main surface in neutral)																																															
Aileron	Up	20°, +1°, -0°	Down 20°, +1°, -0°																																												
Elevator	Up	5°, +1°, -0°	Down 30°, +1°, -0°																																												
Rudder	Right	11°, +1°, -0°	Left 16°, +1°, -0°																																												
(Read degrees normal to rudder hinge line)																																															
Serial Nos. Eligible	414-0001 through 414-0965																																														

VI. Model 421B, Golden Eagle (Normal Category), Approved April 28, 1970

Engines	Two Continental GTSIO-520-H reduction gear ratio .667:1
Fuel	Grade 100 or 100LL aviation gasoline
Engine Limits	For all operations, 2275 propeller r.p.m. (375 hp.) 39.5 in. Hg. Mp. up to critical altitude of 18,000 ft. in standard atmosphere. Above 18,000 ft. the following maximum Mp. applies for maximum r.p.m.:

<u>Altitude (ft.)</u>	<u>Max. Allowable Mp. (in. Hg.)</u>
18,000	39.5
20,000	37.5
22,000	35.5
24,000	33.5
25,000	32.5
26,000	31.3
28,000	28.5
30,000	25.5

VI. Model 421B (cont'd)**Propeller and
Propeller Limits**

- Two McCauley full-feathered 3-bladed propeller installations
- (a) McCauley hub 3AF34C92 with 90LF-0 blades or
McCauley hub 3AF37C516 with 90LFB-0 blades
Diameter: not over 90.0 in., not under 88.0 in.
(no further reduction permitted)
Pitch settings at 30 in. station:
low 16.9°, $\pm 0.2^\circ$
feathering 84.5°, $\pm 0.3^\circ$
- (b) Model 421B S/N 421B0001 thru 421B0500
Hydraulic governor Woodward 210594, 210595, 210596 or 210597
Model 421B S/N 421B0501 and up
McCauley DCF290D2/T4, DFC7290D2/T4, DCFS290D2/T4, DCFUS290D2/T4,
DCF290D7/T4, DCFU290D7/T4, DCFS290D7/T4, DCFUS290D7/T4,
DCFU290D13/T4 or DCFUS290D13/T4.
- (c) Propeller spinner and bulkhead assembly, McCauley D-3534/D-3796.

**Airspeed Limits
(CAS)**

<u>Model 421B: S/N 421B0001 through 421B0500</u>	
Maneuvering	175 m.p.h. (152 knots)
Maximum structural cruising	230 m.p.h. (200 knots)
Never exceed	274 m.p.h. (238 knots)
Landing gear operating	165 m.p.h. (143 knots)
Landing gear extended	165 m.p.h. (143 knots)
Flaps extended 15°	180 m.p.h. (156 knots)
(S/N 421B0001 through 421B0200)	
Flaps extended 15°	200 m.p.h. (174 knots)
(S/N 421B0201 through 421B0500)	
Flaps extended 45°	165 m.p.h. (143 knots)
Minimum control	100 m.p.h. (87 knots)

Model 421B: S/N 421B0501 and On

Maneuvering	152 KCAS (175 m.p.h.)
Maximum structural cruising	200 KCAS (230 m.p.h.)
Never exceed	238 KCAS (274 m.p.h.)
Landing gear operating	145 KCAS (167 m.p.h.)
Landing gear extended	145 KCAS (167 m.p.h.)
Flaps extended 15°	175 KCAS (202 m.p.h.)
Flaps extended 45°	145 KCAS (167 m.p.h.)
Minimum control	87 KCAS (100 m.p.h.)
(S/N 421B0501 through 421B0800)	
Minimum control	82 KCAS (94 m.p.h.)
(S/N 421B0801 and On)	

**C.G. Range (Landing
Gear Extended)**

<u>S/N 421B0001 through 421B0200</u>	
<u>6, 7, or 8 Place</u>	<u>10 Place</u>
(+151.8) to (+156.4) at 7250 lb.	(+151.8) to (+157.7) at 7250 lb.
(+156.7) at 6850 lb. or less	(+158.0) at 6850 lb. or less
(+147.1) at 6100 lb. or less	(+147.1) at 6100 lb. or less
<u>S/N 421B0201 and On</u>	
(+152.6) to (+156.5) at 7450 lb.	(+152.6) to (+157.8) at 7450 lb.
(+156.7) at 7050 lb. or less	(+158.0) at 7050 lb. or less
(+147.1) at 6100 lb. or less	(+147.1) at 6100 lb. or less
Straight line variation between points given	
Landing gear retracted moment change: +889 in.-lb.	

VI. Model 421B (cont'd)

Empty Wt. C.G. Range	None																																																																						
Leveling Means	External screw heads on right side of fuselage at stations +213.9 and +238.55 on W.L. +93.80																																																																						
Maximum Weight	Landing 7200 lb., takeoff 7250 lb. (S/N 421B0001 through 421B0200) Landing 7200 lb., takeoff 7450 lb. (S/N 421B0201 and up)																																																																						
No. of Seats	<u>S/N 421B0001 through 421B0300</u> 6, 7, or 8 (2 at +137.0, 2 at +175.5, 2 at +215.5, 2 at +245.7) or 10 (2 at +137.0, 2 at +161.0, 2 at +190.0, 2 at +218.0, 2 at +249.0) <u>S/N 421B0301 and On</u> 6, 7, or 8 (2 at +137.0, 2 at +175.0, 2 at +218.0, 2 at +261.0) or 10 (2 at +137.0, 2 at +162.0, 2 at +190.0, 2 at +218.0, 2 at +246.0) (See manufacturer's equipment list for optional seating arrangements)																																																																						
Maximum Baggage	<u>S/N 421B0001 through 421B0300</u> 250 lb. (+32.0), 350 lb. (+71.0), 400 lb. (+186.0), 340 lb. (+246.5) <u>S/N 421B0301 and On</u> 250 lb. (+32.0), 350 lb. (+71.0), 400 lb. (+186.0), 400 lb. (+266.0), 100 lb. (+282.0)																																																																						
Fuel Capacity	175 gal. (2 wing tip tanks, 51 gal. ea., 50 gal. usable at +152.0 and 2 wing tanks, 36.5 gal. ea., 35 gal. usable at +164.0) See NOTE 1 for data on unusable fuel																																																																						
Oil Capacity	26 qt. (13 qt. in ea. engine at +115.4; usable 7.0 qt. per engine) See NOTE 1 for data on undrainable oil																																																																						
Control Surface Movements	<table><tr><td>Wing flaps</td><td></td><td></td><td>Down</td><td>45°, +1°, -0°</td></tr><tr><td>Main surfaces</td><td></td><td></td><td></td><td></td></tr><tr><td>Aileron</td><td>Up</td><td>20°, +1°, -0°</td><td>Down</td><td>20°, +1°, -0°</td></tr><tr><td>Elevator</td><td>Up</td><td>25°, +1°, -0°</td><td>Down</td><td>15°, +1°, -0°</td></tr><tr><td>Rudder</td><td>Right</td><td>25°, +1°, -0°</td><td>Left</td><td>25°, +1°, -0°</td></tr><tr><td></td><td></td><td>(S/N 421B0001 through 421B0800)</td><td></td><td></td></tr><tr><td></td><td>Right</td><td>32°, +1°, -0°</td><td>Left</td><td>32°, +1°, -0°</td></tr><tr><td></td><td></td><td>(S/N 421B0801 and On)</td><td></td><td></td></tr><tr><td></td><td></td><td>(Read degrees normal to rudder hinge line)</td><td></td><td></td></tr><tr><td>Tab (main surface in neutral)</td><td></td><td></td><td></td><td></td></tr><tr><td>Aileron</td><td>Up</td><td>20°, +1°, -0°</td><td>Down</td><td>20°, +1°, -0°</td></tr><tr><td>Elevator</td><td>Up</td><td>12°, +1°, -0°</td><td>Down</td><td>20°, +1°, -0°</td></tr><tr><td>Rudder</td><td>Right</td><td>11°, +1°, -0°</td><td>Left</td><td>16°, +1°, -0°</td></tr><tr><td></td><td></td><td>(Read degrees normal to rudder hinge line)</td><td></td><td></td></tr></table>	Wing flaps			Down	45°, +1°, -0°	Main surfaces					Aileron	Up	20°, +1°, -0°	Down	20°, +1°, -0°	Elevator	Up	25°, +1°, -0°	Down	15°, +1°, -0°	Rudder	Right	25°, +1°, -0°	Left	25°, +1°, -0°			(S/N 421B0001 through 421B0800)				Right	32°, +1°, -0°	Left	32°, +1°, -0°			(S/N 421B0801 and On)					(Read degrees normal to rudder hinge line)			Tab (main surface in neutral)					Aileron	Up	20°, +1°, -0°	Down	20°, +1°, -0°	Elevator	Up	12°, +1°, -0°	Down	20°, +1°, -0°	Rudder	Right	11°, +1°, -0°	Left	16°, +1°, -0°			(Read degrees normal to rudder hinge line)		
Wing flaps			Down	45°, +1°, -0°																																																																			
Main surfaces																																																																							
Aileron	Up	20°, +1°, -0°	Down	20°, +1°, -0°																																																																			
Elevator	Up	25°, +1°, -0°	Down	15°, +1°, -0°																																																																			
Rudder	Right	25°, +1°, -0°	Left	25°, +1°, -0°																																																																			
		(S/N 421B0001 through 421B0800)																																																																					
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Tab (main surface in neutral)																																																																							
Aileron	Up	20°, +1°, -0°	Down	20°, +1°, -0°																																																																			
Elevator	Up	12°, +1°, -0°	Down	20°, +1°, -0°																																																																			
Rudder	Right	11°, +1°, -0°	Left	16°, +1°, -0°																																																																			
		(Read degrees normal to rudder hinge line)																																																																					
Serial Nos. Eligible	421B0001 through 421B0970																																																																						

VII. Model 421C, Golden Eagle, (Normal Category), Approved October 28, 1975

Engines	Two Continental GTSIO-520-L reduction gear ratio .667:1 (S/N 421C0001 through 421C1000) Two Continental GTSIO-520-N reduction gear ratio .667:1 (S/N 421C1001 and On)
Fuel	Grade 100 or 100LL aviation gasoline

VII. Model 421C (cont'd)

Engine Limits

For all operations, 2235 propeller r.p.m. (375 hp.)
39.0 in. Hg. Mp. up to critical altitude of 20,000 ft. in standard atmosphere. Above 20,000 ft. the following maximum Mp. applies for maximum r.p.m.:

<u>Altitude (ft.)</u>	<u>Max. Allowable Mp. (in. Hg.)</u>
20,000	39.0
22,000	36.5
24,000	34.0
25,000	32.5
26,000	31.0
28,000	28.0
30,000	25.0

Propeller and
Propeller Limits

Two McCauley full-feathering 3-bladed propeller installations

- (a) McCauley hub 3FF32C501 with 90UMB-0 blades
Diameter: not over 90.0 in., not under 88.0 in.
(no further reduction permitted)
Pitch settings at 30 in. station:
low 16.6°, ±0.2°, feathering 84.6°, ±0.3°
- (b) S/N 421C0001 through 421C0800
Hydraulic Governor McCauley DCF290D2/T6, DCFU290D2/T6,
DCFS290D2/T6, DCFUS290D2/T6, DCF290D7/T6, DCFU290D7/T6 or
DCFU290D13/T6, DCFS290D7/T6, DCFUS290D7/T6 or DCFUS290D13/T6
S/N 421C0801 and On
Hydraulic Governor McCauley DCF290D7/T6, DCFU290D7/T6 or
DCFU290D13/T6, DCFS290D9/T6, DCFUS290D9/T6
- (c) Propeller spinner and bulkhead assembly, McCauley D-3534/D-4506 or McCauley D-5212/D-5217

Airspeed Limits
(IAS)

Maneuvering	151 KIAS (174 m.p.h.)
Maximum structural cruising	201 KIAS (231 m.p.h.)
Never exceed	240 KIAS (276 m.p.h.)
Landing gear operating	176 KIAS (203 m.p.h.)
Landing gear extended	176 KIAS (203 m.p.h.)
Flaps extended 15°	176 KIAS (203 m.p.h.)
Flaps extended 45°	146 KIAS (168 m.p.h.)
Minimum control	80 KIAS (92 m.p.h.)

C.G. Range (Landing
Gear Extended)

6, 7, 8, 9 or 10 Place
(+152.6) to (+158.0) at 7450 lb.
(+147.1) at 6100 lb. or less
Straight line variation between points given
Landing gear retracted moment change:
+917 in.-lb. (S/N 421C0001 through 421C0800)
+1318 in.-lb. (S/N 421C0801 and up)

Empty Wt. C.G. Range

None

Leveling Means

External screw heads on right side of fuselage at stations +213.9 and +238.55
On W.L. +93.80

Maximum Weight

S/N 421C0001 through 421C0400
Landing 7200 lb., takeoff 7450 lb.

S/N 421C0401 and On
Landing 7200 lb., takeoff 7450 lb., ramp 7500 lb.

VII. Model 421C (cont'd)

No. of Seats	6, 7 or 8 (2 at +137.0, 2 at +175.0, 2 at +218.0, 1 at +261.0) or 10 (2 at +137.0, 2 at +162.0, 2 at +190.0, 2 at +218.0, 2 at +246.0) (See manufacturer's equipment list for optional seating arrangements)		
Maximum Baggage	250 lb. (+32.0), 350 lb. (+71.0), 400 lb. (+186.0), 400 lb. (+266.0), 100 lb. (+282.0)		
Fuel Capacity	213.4 gal. (2 wing tanks, 106.7 gal. ea., 103.0 gal. usable at +161.0) See NOTE 1 for data on unusable fuel		
Oil Capacity	26 qt. (13 qt. in ea. engine at +115.4; usable 7.0 qt. per engine) See NOTE 1 for data on undrainable oil		
Control Surface Movements	Wing flaps		Down 45°, +1°, -0°
	Main surfaces		
	Aileron	Up 20°, +1°, -0°	Down 20°, +1°, -0°
	Elevator	Up 25°, +1°, -0°	Down 15°, +1°, -0°
	Rudder	Right 32°, +1°, -0°	Left 32°, +1°, -0°
	(Read degrees normal to rudder hinge line)		
	Tab (main surface in neutral)		
	Aileron	Up 20°, +1°, -0°	Down 20°, +1°, -0°
	Elevator	Up 12°, +1°, -0°	Down 20°, +1°, -0°
	Rudder	Right 11°, +1°, -0°	Left 16°, +1°, -0°
	(Read degrees normal to rudder hinge line)		
Serial Nos. Eligible	421C0001 through 421C1807		

VIII. Model 414A, Chancellor, (Normal Category), Approved September 30, 1977

Engines	Two Continental TSIO-520-N or TSIO-520-NB (in any combination) (S/N 414A0001 through 414A0200)
	Two Continental TSIO-520-NB (S/N 414A0201 and On)
Fuel	Grade 100 or 100LL Aviation Gasoline
Engine Limits	For all operations, 2700 r.p.m., 310 hp., 38.0 in. Hg. Mp. up to critical altitude of 20,000 ft. in standard atmosphere. Above 20,000 ft. the following maximum Mp. applies for maximum r.p.m.:

<u>Altitude (ft.)</u>	<u>Max. Allowable Mp. (in. Hg.)</u>
20,000	38.0
22,000	35.2
24,000	32.3
26,000	29.8
28,000	27.4
30,000	25.0

Propeller and Propeller Limits	Two McCauley full-feathering three-bladed propeller installations
	(a) McCauley hub 3AF32C93 with 82NC-5.5 blades or McCauley hub 3AF32C505 with 82NEA-5.5 blades Diameter: not over 76.5 in., not under 75.0 in. (no further reduction permitted) Pitch settings at 30 in. station: low 14.9°, ±0.2°, feathering 81.2°, ±0.3°
	or (b) McCauley hub 3AF32C93 with 82NC-5.5 blades or McCauley hub 3AF32C505 with 82NEA-5.5 blades Diameter: not over 75.5 in., not under 75 in. Pitch settings at 30 in. station: low 15.2°, ±0.2° feathered 81.2°, ±0.3°

VIII. Model 414A (cont'd)

Propeller and Propeller Limits (cont'd)	<p>(c) <u>S/N 414A0001 through 414A0801</u> Hydraulic governor McCauley DCF290D2/T3, DCFU290D2/T3, DCFS290D4/T3, DCFUS290D4/T3, DCFS290D6/T3, DCFUS290D6/T3, DCF290D7/T3, DCFU290D7/T3, DCFU290D13/T3, DCFS290D7/T3, DCFUS290D7/T3, DCFUS290D13/T3, DCFS290D8/T3, DCFUS290D8/T3 or DCFUS290D12/T13</p> <p><u>S/N 414AC0801 and On</u> Hydraulic governor McCauley DCF290D2/T3, DCFU290D2/T3, DCF290D7/T3, DCFU290D7/T3 or DCFU290D13/T3, DCFS290D9/T3, DCFUS290D9/T3</p> <p>(d) Propeller spinner and bulkhead assembly, McCauley D-3534/D-3796, or McCauley D-5212/D-5214</p>	
Airspeed Limits (IAS)	<p>Maneuvering 145 KIAS (167 m.p.h.)</p> <p>Max. structural cruising 203 KIAS (234 m.p.h.)</p> <p>Never exceed 237 KIAS (273 m.p.h.)</p> <p>Landing gear operating 177 KIAS (204 m.p.h.)</p> <p>Landing gear extended 177 KIAS (204 m.p.h.)</p> <p>Flaps extended 15° 177 KIAS (204 m.p.h.)</p> <p>Flaps extended 45° 146 KIAS (168 m.p.h.)</p> <p>Minimum control 79 KIAS (91 m.p.h.)</p>	
C.G. Range (Landing Gear Extended)	<p>(+151.3) to (+160.0) at 6750 lb.</p> <p>(+147.8) at 5800 lb. or less</p> <p>Straight line variation between points given</p> <p>Landing gear retracted moment change: +917 in.-lb.</p>	
Empty Wt. C.G. Range	None	
Leveling Means	External screw heads on right side of fuselage at stations +213.29 and +238.55 on W.L. +93.80	
Maximum Weight	Ramp 6785 lb., takeoff and landing 6750 lb.	
No. of Seats	6, 7 or 8 (2 at +137.0, 2 at +175.0, 2 at +218.0, Optional: 1 or 2 at +261.0 or with toilet option, 1 at +250.0) (See manufacturer's equipment list for optional seating arrangements)	
Maximum Baggage	250 lb. (+32.0), 350 lb. (+71.0), 400 lb. (+186.0), 400 lb. (+266.0), 100 lb. (+282.0)	
Fuel Capacity	<p><u>S/N 414A0001 through 414A0200</u> 213.4 gal. (2 wing tanks, 106.7 gal. ea., 103.0 gal. usable at +161.0) See NOTE 1 for data on unusable fuel</p> <p><u>S/N 414A0201 through 414A0400</u> 213.4 gal. (2 wing tanks, 106.7 gal. ea., 102.0 gal. usable at +161.0) See NOTE 1 for data on unusable fuel</p> <p><u>S/N 414A0401 and On</u> 213.4 gal. (2 wing tanks, 106.7 gal. ea., 103.0 gal. usable at +161.0) See NOTE 1 for data on unusable fuel</p>	
Oil Capacity	26 qt. (13 qt. in ea. engine at +110.9; usable 6.5 qt. per engine) See NOTE 1 for data on undrainable oil	

VIII. Model 414A (cont'd)

Control Surface Movements	Wing flaps	Down	45°, +1°, -0°
	Main surfaces		
	Aileron	Up	20°, +1°, -0°
	Elevator	Up	25°, +1°, -0°
	Rudder	Right	32°, +1°, -0°
		Left	32°, +1°, -0°
	(Read degrees normal to rudder hinge line)		
	Tab (main surface in neutral)		
	Aileron	Up	20°, +1°, -0°
	Elevator	Up	12°, +1°, -0°
	Rudder	Right	11°, +1°, -0°
		Left	16°, +1°, -0°
	(Read degrees normal to rudder hinge line)		

Serial Nos. Eligible 414A0001 through 414A1212

IX. Model 402C, Business Liner/Utility Liner, (Normal Category), Approved September 25, 1978

Engines Two Continental TSIO-520-VB rated at 325 hp.

Fuel Grade 100 or 100LL aviation gasoline

Engine Limits Takeoff and engine inoperative, 2700 r.p.m., 39.0 in. Hg. Mp. up to 12,000 ft. Above 12,000 ft. the following maximum Mp. applies for maximum r.p.m.

<u>Altitude (ft.)</u>	<u>Max. Allowable Mp. (in. Hg.)</u>
S.L. to	
12,000	39.0
14,000	37.2
16,000	37.2
18,000	32.0
20,000	29.5
22,000	27.0
24,000	25.0
26,000	23.0
28,000	21.0
30,000	19.0

Propeller and
Propeller Limits Two McCauley full-feathering three-bladed propeller installations

(a) McCauley hub 3AF32C93 with 82NC-5.5 blades or McCauley hub 3AF32C505 with 82NEA-5.5 blades
Diameter: not over 76.5 in., not under 75.0 in.
(no further reduction permitted)
Pitch settings at 30 in. station:
low 14.9°, ±0.2°, feathering 82.2°, ±0.3°

or (b) McCauley hub 3AF32C93 with 82NC-6.5 blades or McCauley hub 3AF32C505 with 82NEA-6.5 blades
Diameter: not over 75.5 in., not under 75.0 in.
Pitch settings at 30 in. station:
low 15.2°, ±0.2°, feathering 82.2°, ±0.3°

(c) S/N 402C0001 through 402C0600
Hydraulic governor, Woodward B210444, C210439; McCauley DCF290D7/T3, DCFUS290D7/T3, DCFU290D13/T3, DCFS290D7/T3, DCFUS290D7/T3, DCFUS290D13/T3, DCFUS290D8/T3, or DCFUS290D12/T3
S/N 689, and 402C0601 and On
Hydraulic governor, Woodward B210444, C210439; McCauley DCF290D7/T3, DCFU290D7/T3 or DCFU290D13/T3, DCFS290D9/T3, DCFUS290D9/T3

(d) Propeller spinner and bulkhead assembly; McCauley D-3534/D-3537, D-3534/D-3796, or D-5212/D-5214

IX. Model 402C (cont'd)

Airspeed Limits (IAS)	Maneuvering	150 KIAS (173 m.p.h.)
	Max. structural cruising	205 KIAS (236 m.p.h.)
	Never exceed	235 KIAS (270 m.p.h.)
	Landing gear operating	180 KIAS (207 m.p.h.)
	Landing gear extended	180 KIAS (207 m.p.h.)
	Flaps extended 15°	180 KIAS (207 m.p.h.)
	Flaps extended 45°	149 KIAS (172 m.p.h.)
	Minimum control	80 KIAS (92 m.p.h.)
C.G. Range (Landing Gear Extended)	(+151.58) to (+160.67) at 6850 lb. (+149.08) at 5800 lbs. or less Straight line variation between points given Landing gear retracted moment change: +917 in.-lb.	
Empty Wt. C.G. Range	None	
Leveling Means	External screw heads on right side of fuselage at stations +213.65 and +238.00 on W.L. +93.80	
Maximum Weight	Ramp, 6885 lbs., takeoff and landing 6850 lbs.	
No. of Seats	6, 7 or 8 (2 at +137.0, 2 at +175.0, 2 at +218.0, 1 or 2 at +261.0) 9 (with photographic provisions option) (2 at +137.0, 2 at +162.0, 2 at +190.0, 2 at +218.0, 1 at +246.0) 10 (2 at +137.0, 2 at +162.0, 2 at +190.0, 2 at +218.0, 2 at +246.0) (See manufacturer's equipment list for optional seating arrangements)	
Maximum Baggage	250 lbs. (+32.0), 350 lbs. (+71.0), 400 lbs. (+186.0), 400 lbs. (+266.0), 100 lbs. (+282.0)	
Fuel Capacity	<u>S/N 402C0001 through 402C0200</u> 213.4 gal. (2 wing tanks, 106.7 gal. ea., 102 gal. usable at +161.0) See NOTE 1 for data on unusable fuel <u>S/N 689, and 402C0201 and On</u> 213.4 gal. (2 wing tanks, 106.7 gal. ea., 103 gal. usable at +161.0) See NOTE 1 for data on unusable fuel	
Oil Capacity	26 qt. (13 qt. in ea. engine at +110.9; usable 6.5 qt. per engine) See NOTE 1 for data on undrainable oil	
Control Surface Movements	Wing flaps	Down 45°, +1°, -0°
	Main surfaces	
	Aileron Up 20°, +1°, -0°	Down 20°, +1°, -0°
	Elevator Up 25°, +1°, -0°	Down 15°, +1°, -0°
	Rudder Right 32°, +1°, -0°	Left 32°, +1°, -0°
	(Read degrees normal to rudder hinge line)	
	Tab (main surface in neutral)	
	Aileron Up 20°, +1°, -0°	Down 20°, +1°, -0°
	Elevator Up 12°, +1°, -0°	Down 20°, +1°, -0°
	Rudder Right 11°, +1°, -0°	Left 16°, +1°, -0°
	(Read degrees normal to rudder hinge line)	
Serial Nos. Eligible	689, 402C0001 through 402C1020	

X. Model 425, Corsair or Conquest I (See NOTE 7), (Normal Category), Approved July 1, 1980

Engines Two Pratt & Whitney Aircraft of Canada, Ltd., PT6A-112 turboprop

Fuel Aviation turbine fuel Jet A, Jet A-1, or Jet B, JP-4, JP-5 or JP-8. For required use of anti-icing additives and emergency use of aviation gasoline, refer to the Pilot's Operating Handbook and FAA Approved Airplane Flight Manual.

Engine Limits	Operating Limits				
	Shaft Horsepower Power	Ng Gas Generator Speed (% rpm)	Indicated Torque (ft.-lbs.)	Prop. Shaft Speed (rpm)	Maximum Permissible Interturbine Temp. (°C.)
Takeoff static & max. continuous	450*	101.6	1244	1900	725
Starting (2 sec.)	--	--	--	--	1090
Maximum reverse	430	101.6	1244	1815	725

*Flat Rated:

The engines may produce more power than that for which the airplane has been certificated.

Under these conditions, the placarded torque meter, ITT, or Ng limitations shall not be exceeded.

- Propeller and
Propeller Limits**
- (1) Two Hartzell three-bladed, full-feathered, reversible
 - Hub: HC-B3TN-3C
 - Blade: T10178B-8R
 - Diameter: Not over 93-3/8 in., not under 91 inches; no further reduction permitted
 - Pitch at 30-inch station:
 - Low pitch 20.2°
 - Feathered 86.7°
 - Reverse -10.9°
 - (2) Two McCauley three-bladed, full-feathered, reversible
 - Hub: 3GFR34C701
 - Blade: 93KB-0
 - Diameter: Not over 93 inches, not under 90-5/8 inches; no further reduction permitted
 - Pitch at 30-inch station:
 - Low pitch 18.5°
 - Feathered 85.5°
 - Reverse -13.5°

Propellers may be interchanged in any combination.

Airspeed Limits (IAS)	V _{MO} (Max Operating)	230 knots 265 m.p.h.
	Sea level to 21,800 ft.	
	M _{MO} Above 21,800 ft.	.52 mach
	V _A (Maneuvering) at 8200 lbs.	154 knots 177 m.p.h.
	V _A (Maneuvering) at 8600 lbs.	157 knots 181 m.p.h.
	V _{FE} (Flaps extended)	
	45° (Landing)	145 knots 169 m.p.h.
	15° (Takeoff & Approach)	175 knots 201 m.p.h.
	V _{MCA} (Min. control speed) Air at 8200 lbs.	90 knots 104 m.p.h.
	V _{MCA} (Min. control speed) Air at 8600 lbs.	92 knots 106 m.p.h.
	V _{LE} (Landing gear extended)	175 knots 201 m.p.h.

X. Model 425 (cont'd)

C.G. Range (Landing Gear Extended)	<u>S/N 425-0001 through 425-0176 (See NOTE 7)</u> (155.66) to (160.04) at 8200 lbs. (150.65) to (160.04) at 6478 lbs. or less <u>S/N 425-0177 and On</u> (156.81) to (160.04) at 8600 lbs. (150.65) to (160.04) at 6478 lbs. or less Straight line variation between points given Moment change due to retracting landing gear (+1448 in.-lb.)																																																										
Empty Wt. C.G. Range	None																																																										
Leveling Means	External screw heads on right side of fuselage at stations +213.9 and +238.55 on W.L. +93.80																																																										
Maximum Weight	<u>S/N 425-0001 through 425-0176</u> <u>(See NOTE 7)</u> Takeoff 8200 lbs. Landing 8000 lbs. Zero fuel 6740 lbs. Ramp 8275 lbs.		<u>S/N 425-0177 and On</u> 8600 lbs. 8000 lbs. 7000 lbs. 8675 lbs.																																																								
No. of Seats	6, 7 or 8 (2 at +137.0, 2 at +175.0, 2 at +218.0, 2 at +261.0) See manufacturer's equipment list for optional seating arrangements																																																										
Maximum Baggage	250 lb. (+32.0), 350 lb. (+71.0), 400 lb. (+266.0), 100 lb. (+282.0)																																																										
Fuel Capacity	2497.8 lb. (372.8 gal.) total in two wing tanks, 1248.9 lb. (186.4 gal.) each; 2452.2 lb. (366.0 gal.) usable total, 1226.1 lb. (133 gal.) in each tank at +163.3. Fuel weight based on 6.70 lb./gal. See NOTE 1 for data on unusable fuel.																																																										
Oil Capacity	5.28 gal. total, 5.28 gal. usable (2.3 gal. in each engine-mounted tank at +125.3). See NOTE 1 for data on undrainable oil.																																																										
Maximum Operating Altitude	30,000 ft.																																																										
Control Surface Movements	<table><tr><td colspan="3">Wing flaps</td><td>Down</td><td>45°, +1°, -0°</td></tr><tr><td colspan="5">Main surfaces</td></tr><tr><td>Aileron</td><td>Up</td><td>20°, +1°, -0°</td><td>Down</td><td>20°, +1°, -0°</td></tr><tr><td>Elevator</td><td>Up</td><td>19°, +1°, -0°</td><td>Down</td><td>15°, +1°, -0°</td></tr><tr><td>Rudder</td><td>Right</td><td>32°, +1°, -0°</td><td>Left</td><td>32°, +1°, -0°</td></tr><tr><td colspan="5">(Read degrees normal to rudder hinge line)</td></tr><tr><td colspan="5">Tab (main surface in neutral)</td></tr><tr><td>Aileron</td><td>Up</td><td>20°, +1°, -0°</td><td>Down</td><td>20°, +1°, -0°</td></tr><tr><td>Elevator</td><td>Up</td><td>6°, +1°, -0°</td><td>Down</td><td>15°, +1°, -0°</td></tr><tr><td>Rudder</td><td>Right</td><td>11°, +1°, -0°</td><td>Left</td><td>16°, +1°, -0°</td></tr><tr><td colspan="5">(Read degrees normal to rudder hinge line)</td></tr></table>				Wing flaps			Down	45°, +1°, -0°	Main surfaces					Aileron	Up	20°, +1°, -0°	Down	20°, +1°, -0°	Elevator	Up	19°, +1°, -0°	Down	15°, +1°, -0°	Rudder	Right	32°, +1°, -0°	Left	32°, +1°, -0°	(Read degrees normal to rudder hinge line)					Tab (main surface in neutral)					Aileron	Up	20°, +1°, -0°	Down	20°, +1°, -0°	Elevator	Up	6°, +1°, -0°	Down	15°, +1°, -0°	Rudder	Right	11°, +1°, -0°	Left	16°, +1°, -0°	(Read degrees normal to rudder hinge line)				
Wing flaps			Down	45°, +1°, -0°																																																							
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Rudder	Right	11°, +1°, -0°	Left	16°, +1°, -0°																																																							
(Read degrees normal to rudder hinge line)																																																											
Serial Nos. Eligible	425-0001 through 425-0236																																																										

Data Pertinent to All Models

Datum 100.00 in. forward face of fuselage bulkhead forward of rudder pedals.

Certification Basis:

Models 401, 401A, 401B, 402, 402A, 402B, 411, 411A, 414, 421, 421A:

Part 3 of the Civil Air Regulations dated May 15, 1956, as amended by 3-1 through 3-5 and 3-8.

Model 421B:

Part 3 of the Civil Air Regulations dated May 15, 1956, except Subpart B, as amended by 3-1 through 3-5 and 3-8; Subpart B, paragraphs 23.25 through 23.253 of the Federal Aviation Regulations dated February 1, 1965, as amended by 23-1 through 23-7.

Models 414A, 421C:

Part 3 of the Civil Air Regulations dated May 15, 1956, as amended by 3-1 through 3-5 and 3-8, excluding the following portions: Subpart B and paragraphs 3.356, 3.357, 3.358, 3.359, 3.411, 3.429, 3.433, 3.434, 3.435, 3.436, 3.437, 3.445, 3.581, 3.582, 3.583, 3.584, 3.585, 3.587, 3.628, 3.666, 3.672, 3.673, 3.674, 3.675, 3.700(c), 3.728, 3.767(a) and 3.767(b). Include the following portions of FAR 23 dated February 1, 1965, as amended by 23-1 through 23-14; Subpart B and paragraphs 23.729, 23.901, 23.909, 23.951, 23.954, 23.955, 23.959, 23.973, 23.1041, 23.1043, 23.1047, 23.1143, 23.1305, 23.1387(e), 23.1435 and 23.1557(c); as amended by 23-1 through 23-21, paragraph 23.1385(c); as amended by 23-1 through 23-23, paragraph 23.1327. Add paragraph 23.1559(b) for Model 414A only. Findings of Equivalent Level of Safety were made for CAR 3.637, 3.757, and 3.778(a).

Model 402C:

Part 3 of the Civil Air Regulations dated May 15, 1956, as amended by 3-1 through 3-5 and 3-8, excluding the following portions: Subpart B and paragraphs 3.356, 3.357, 3.358, 3.359, 3.411, 3.429, 3.433, 3.434, 3.435, 3.436, 3.437, 3.445, 3.581, 3.582, 3.583, 3.584, 3.585, 3.587, 3.628, 3.666, 3.672, 3.673, 3.674, 3.675, 3.700(c), 3.728, 3.767(a) and 3.767(b). Include the following portions of FAR 23 dated February 1, 1965, as amended by 23-1 through 23-14: Subpart B and paragraphs 23.729, 23.901, 23.909, 23.951, 23.954, 23.955, 23.959, 23.973, 23.1041, 23.1043, 23.1047, 23.1143, 23.1305, 23.1387(e), 23.1435, 23.1557(c), and 23.1559(b); as amended by 23-1 through 23-21, paragraph 23.1385(c); as amended by 23-1 through 23-23, paragraph 23.1327.

Part 36 of the Federal Aviation Regulations dated December 1, 1969, as amended by 36-1 through 36-7. Findings of Equivalent Level of Safety were made for CAR 3.637, 3.757, and 3.778(a).

Model 425:

Part 3 of the Civil Air Regulations dated May 15, 1956, as amended by 3-1 through 3-6 and 3-8 as follows: Paragraphs 3.0 through 3.20, 3.291 through 3.307, 3.317 through 3.347, 3.371 through 3.401, 3.651, 3.652, 3.655(c) and (d), 3.661, 3.662, 3.668, 3.686 through 3.699, 3.711 through 3.728, 3.749, 3.791, and 3.792; the following portions of FAR 23 dated February 1, 1965, as amended by 23-1 through 23-21: Paragraphs 23.21 through 23.33, 23.45(a) through (d), 23.49 through 23.179, 23.181(a), 23.201 through 23.572, 23.629, 23.723 through 23.735, 23.865, 23.867, 23.901 through 23.1017, 23.1019(a)(1) and (2), 23.1019(a)(4) and (5), 23.1019(b), 23.1021 through 23.1203, 23.1303(a) through (d), 23.1305(a) through (u) and (w), 23.1323, 23.1325, 23.1327, 23.1329, 23.1335, 23.1337, 23.1351 through 23.1357, 23.1385 through 23.1401, 23.1441 through 23.1449, 23.1501 through 23.1521, 23.1524, 23.1525, 23.1527(b), and 23.1529 through 23.1589; Paragraph 25.831(d) of FAR 25 dated February 1, 1965, as amended by 25-1 through 25-43; FAR 36 dated December 1, 1969, as amended by 36-1 through 36-10; SFAR No. 27, Fuel Venting and Exhaust Emission Requirements for Turbine Engine Powered Airplanes, effective February 1, 1974, as amended by SFAR's 27-1, 27-2, and 27-3; plus Special Conditions 23-93-CE-12 as amended by Amendment No. 1 dated June 25, 1980. (See NOTE 3.)

Model 414A (S/N 414A0401 and On), Model 421C (S/N 421C0801 and On)

In addition to the above certification basis, compliance with FAR 36, dated December 1, 1969, as amended by 36-1 through 36-10 (414A only) and 36-1 through 36-4 (421C only) has been demonstrated.

Model 402B (S/N 402B0501 and On), Model 402C, Model 414 (S/N 414-0451 and On), Model 414A, Model 421B (S/N 421B0501 and On), Model 421C, Model 425

Markings, placards and manuals are primarily in knots instead of m.p.h. as required by CAR 3, but permitted by FAR 23, Amendment 23-7.

Model 402B (S/N 402B1001 and On), Model 414 (S/N 414-0801 and On)

Findings of equivalent level of safety were made for CAR 3.757 and 3.778(a).

Data Pertinent to All Models (cont'd)**Certification Basis** (cont'd)

Model 402B (S/N 402B0801 and On), Model 402C, Model 414 (S/N 414-0601 and On), Model 414A, Model 421B (S/N 421B0801 and On), Model 421C, Model 425

In addition to the above certification basis, compliance with ice protection has been demonstrated in accordance with FAR 23.1419 of Amendment 23-14 effective December 20, 1973, when ice protection equipment is installed in accordance with Cessna Drawing 5914105 for 425, 5114400 for all other models, Factory Kit (FK) No. 194, Pilot's Operating Handbook and/or FAA Approved Airplane Flight Manual. Aircraft which have been modified in compliance with Accessory Kit (AK) No. 421-106 are considered to be equivalent to those with Factory Kit (FK) No. 194.

Application for Type Certificate dated September 18, 1961. Type Certificate No. A7CE issued August 17, 1964, obtained by the manufacturer under delegation option procedures.

Production Basis:

Production Certificate No. 312 issued and Delegation Option Manufacturer No. CE-3 authorized to issue airworthiness certificates under delegation option provisions of Part 21 of the Federal Aviation Regulations. Effective February 15, 1985, and on, Production Certificate No. 4 is applicable to all spares production. See NOTE 8 for specific effectivity of P.C. 4 on new airplane serials.

Equipment:

The basic required equipment as prescribed in the applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. In addition, the following item of equipment is required.

- Stall warning indicator, Cessna dwg. 5018100 (401, 402, 411, 411A)
- Stall warning indicator, Cessna dwg. 5118000 (421)
- Stall warning indicator, Cessna dwg. 5618002 (414)
- Stall warning indicator, Cessna dwg. 5218016 (401A, 402A, 401B, 402B0001 through 402B0300)
- Stall warning indicator, Cessna dwg. 5118310 (421A)
- Stall warning indicator, Cessna dwg. 5118402 (421B0001 through 421B0300)
- Stall warning indicator, Cessna dwg. 5618021 (414-0351 and On, 421B0301 and On)
- Stall warning indicator, Cessna dwg. 5218031 (402B0301 and On)
- Stall warning indicator, Cessna dwg. 5118627 (421C)
- Stall warning indicator, Cessna dwg. 5618041 (402C, 414A, 425)
- Or Angle of Attack Indicator System, Cessna Dwg. 0800302, Model 402B, 402C, 414, 414A, 421B, 421C.

NOTES:

NOTE 1. Current weight and balance report together with list of equipment included in certificated empty weight and loading instructions when necessary must be provided for each aircraft at the time of original certification.

The certificated empty weight and corresponding center of gravity location must include undrainable oil (not included in oil capacity) and unusable fuel as follows:

- (a) Fuel. 12 lb. (tip) at (+152.0) (401, 401A, 401B, 402, 402A, 402B, 411, 411A, 414, 421, 421A, 421B)
- 18 lb. (wing, standard 73 gal. at +164.0) (411, 411A, 421, 421A, 421B)
- 24 lb. (wing, optional 100 gal. at +164.0) (411, 411A, 421, 421A, 421B, 402A, 402B, 414)
- 6 lb. (wing, optional 63 gal. at +164.0) (402B0301 and up and 414-0351 and up)
- 44 lb. (wing, 7.4 gal. at +165.2) (402C, S/N 689, and 402C0201 and up; 414A, S/N 414A0401 and up; 421C)
- 68 lb. (wing, 11.4 gal. at +165.2) (414A, S/N 414A0001 through S/N 414A0200)
- 56 lb. (wing, 9.4 gal. at +165.0) (402C, S/N 402C0001 through 402C0200; 414A, S/N 414A0201 through 414A0400)
- 45.6 lb. (wing, 6.8 gal. at +166.2) (425)
- (b) If optional wing locker transfer tanks are installed 3.0 lb. (each 26 gal. tank) at (+176.0) (411, 411A, 421, 421A, 421B)
- 3.0 lb. (each 20 gal. tank) at (+175.0) (401, 401A, 401B, 402, 402A, 402B, 414)
- 2.0 lb. (each 28 gal. tank) at (+176.0) (421C0001 and up)
- (c) Oil - 0.0 lb.

NOTE 2. The placards specified in the FAA Approved Airplane Flight Manual must be displayed.

Data Pertinent to All Models (cont'd)

NOTE 3. Service information
The appropriate airplane service manual contains structural retirement lives, which may not be changed without FAA Engineering approval, for the following components:

	<u>Part Number</u>	<u>Hours</u>	<u>Model</u>
Windshield	5111604-1 & -2 5111604-200 & -201 5111604-3 & -4 5111604-202 & -203	13,200	414, 414A, 421A, 421B, 421C 425
Windshield, heated	9910013-1	13,200	421, 421A (S/N 421A0001 through 421A0117)
Windshield, heated	9910071-1 9910071-200	13,200	414, 421A, 421B (S/N 414-0001 through 414-0600, 421A0118 through 421B0800)
Windshield, heated	9910214-1 & -2 9910214-200	13,200	414, 414A, 421B, 421C (S/N 414-0601 and On, 421B0801 through 421C0800)
Windshield, heated	9910460-1 & -200	13,200	421C (S/N 421C0801 and On), 425
Upper cabin door latch pins	5111545-3	8,000	421 (S/N 421-0001 through 421-0079)
Upper cabin door latch pins	5111545-6	8,000	421 (S/N 421-0080 and On), 421A
Wing	5922125 not modified by SK425-48	10,200	425 (S/N -0002 thru -0176 except airplanes incorporating SK425-17)
Wing	5922125 not modified by SK425-48	9,300	425 (S/N -0177 and On and airplanes -0002 thru -0176 incorporating SK425-17)
Wing	5922125 modified by SK425-48	30,000	425
Wing carry-thru	5911004, 5111225	30,000	425
Wing, lower carry- thru main spar	5211178-1, 5311027-8	12,500	425

For Model 425 aircraft that have exceeded the structural retirement life prior to the availability of Cessna Service Kit SK425-48, the service kit is to be installed according to the following schedule:

A. For airplanes 425-0177 and On, and airplanes 425-0002 through 425-0176 incorporating SK425-17:

Exceeding 12, 500 hours, accomplish SK425-48 within 100 hours or 12 months after SK 425-48 was issued, whichever comes first.

Exceeding 9,300 hours but less than 12,500 hours, accomplish SK425-48 within 400 hours or 24 months after SK425-48 was issued whichever comes first.

Between 8,900 and 9,300 hours when SK425-48 was issued, accomplish within 400 hours of operation. For airplanes with less than 8,900 hours when SK425-48 was issued, accomplish at 9,300 hours.

B. For airplanes -0002 through -0176, except airplanes incorporating SK425-17:

Exceeding 12,500 hours, accomplish SK425-48 within 100 hours or 12 months after SK425-48 was issued whichever comes first.

Exceeding 10, 200 hours but less than 12,500 hours, accomplish SK425-48 within 400 hours or 24 months after SK425-48 was issued whichever comes first.

Between 9,800 and 10,200 hours when SK425-48 was issued, accomplish within 400 hours of operation. For airplanes with less than 9,800 hours when SK425-48 was issued, accomplish 10,200 hours.

Data Pertinent to All Models (cont'd)

NOTE 3. (cont'd.)

Model 425 Special Conditions 23-93-CE-12, required, in part, that Cessna establish mandatory inspections of the Horizontal Tail Assembly in order to maintain continued structural integrity. Therefore, inspections are required for the horizontal stabilizer, elevators, elevator tab and tab actuator system. In order to comply with these requirements, airplanes must be inspected in accordance with inspection Item Codes A273002, A273101, A273102, B273109 and A551001 as contained in Model 425 Maintenance Manual, Part Number D2535-3-13, Revision 3 (or later revision). These inspection criteria are contained in Chapter 5, Subsection 5-10-01, and are applicable to Zones 331 and 332. All approved airplane inspection programs must include these mandatory inspections.

NOTE 4. Model 421, S/N 421-0001 and On, approved for 6840 lb. takeoff weight with C.G. range as follows when appropriate airplane flight manual, pilot's checklist, weight and balance form, and other documents are provided as specified in Cessna Service Kit SK421-12.

[illegible]

Straight line variation between points given

NOTE 5. McCauley propellers with 3AF32C87 and 3AF32C504 hubs may be interchanged in any combination. This also applies to propellers with 3AF32C93 and 3AF32C505m hubs; 3AF34C92 and 3AF37C516 hubs; 3AF34C74 and 3AF37C510 hubs.

NOTE 6. Model 425 aircraft in compliance with Cessna Drawing 5700018 are eligible for certification in The Netherlands.

NOTE 7. Model 425, S/N 425-0001 through 425-0176 (Corsair) are eligible for the maximum weights and C.G. range applicable to S/N 425-0177 and On (Conquest I), when modified in accordance with Cessna Service Kit SK425-17, and will be renamed Conquest I.

NOTE 8. Production Certificate No. 4 effective at Serials 402C1005 and On, 414A1208 and On, 421C1801 and On, and 425-0228 and On.

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