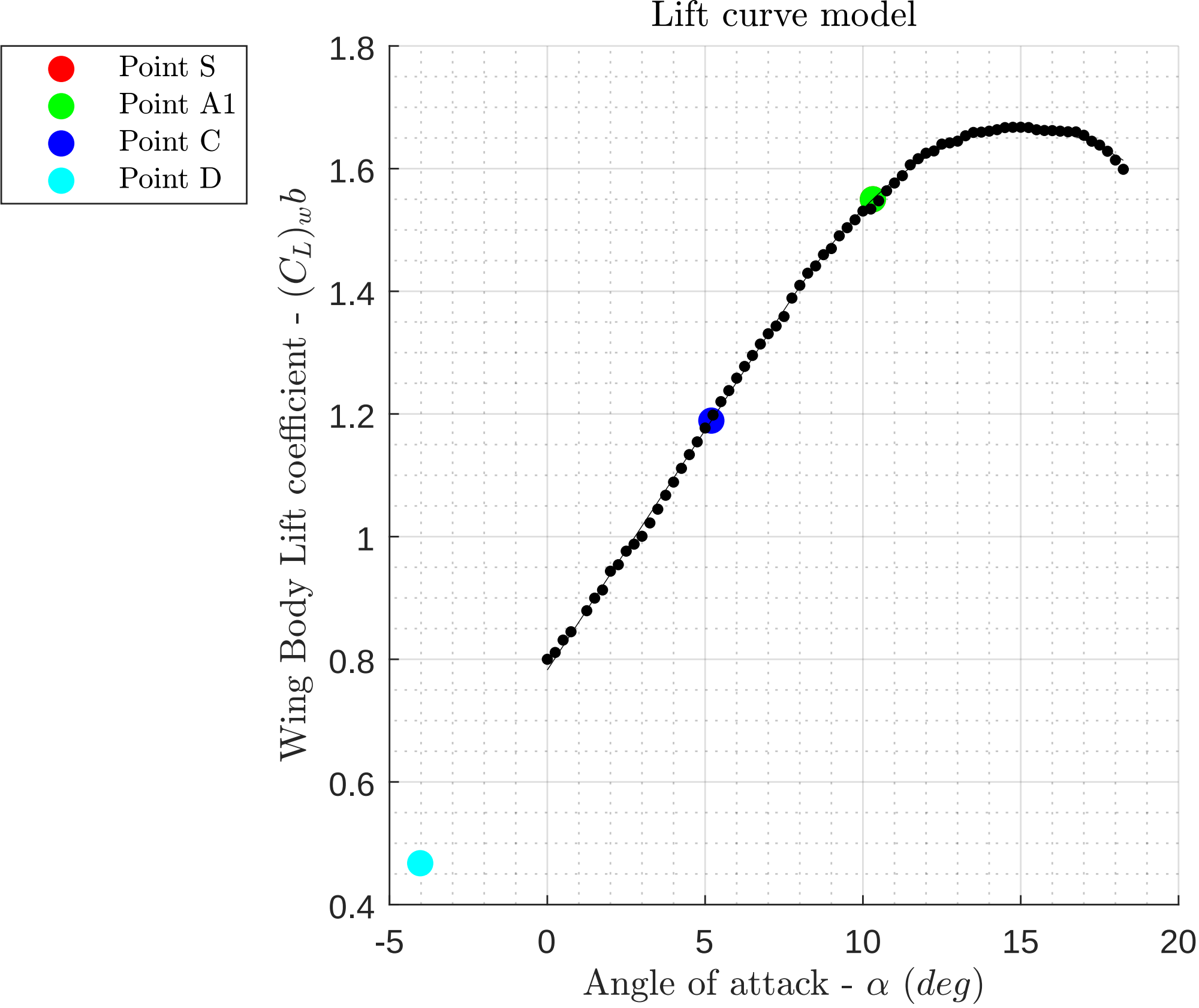
Lift curve model
Point S
Point A1
Point C
Point D
Wing Body Lift coefficient - $(C_L)_w b$
Angle of attack - α (deg)
1.8
1.6
1.4
1.2
1
0.8
0.6
0.4
-5
0
5
10
15
20