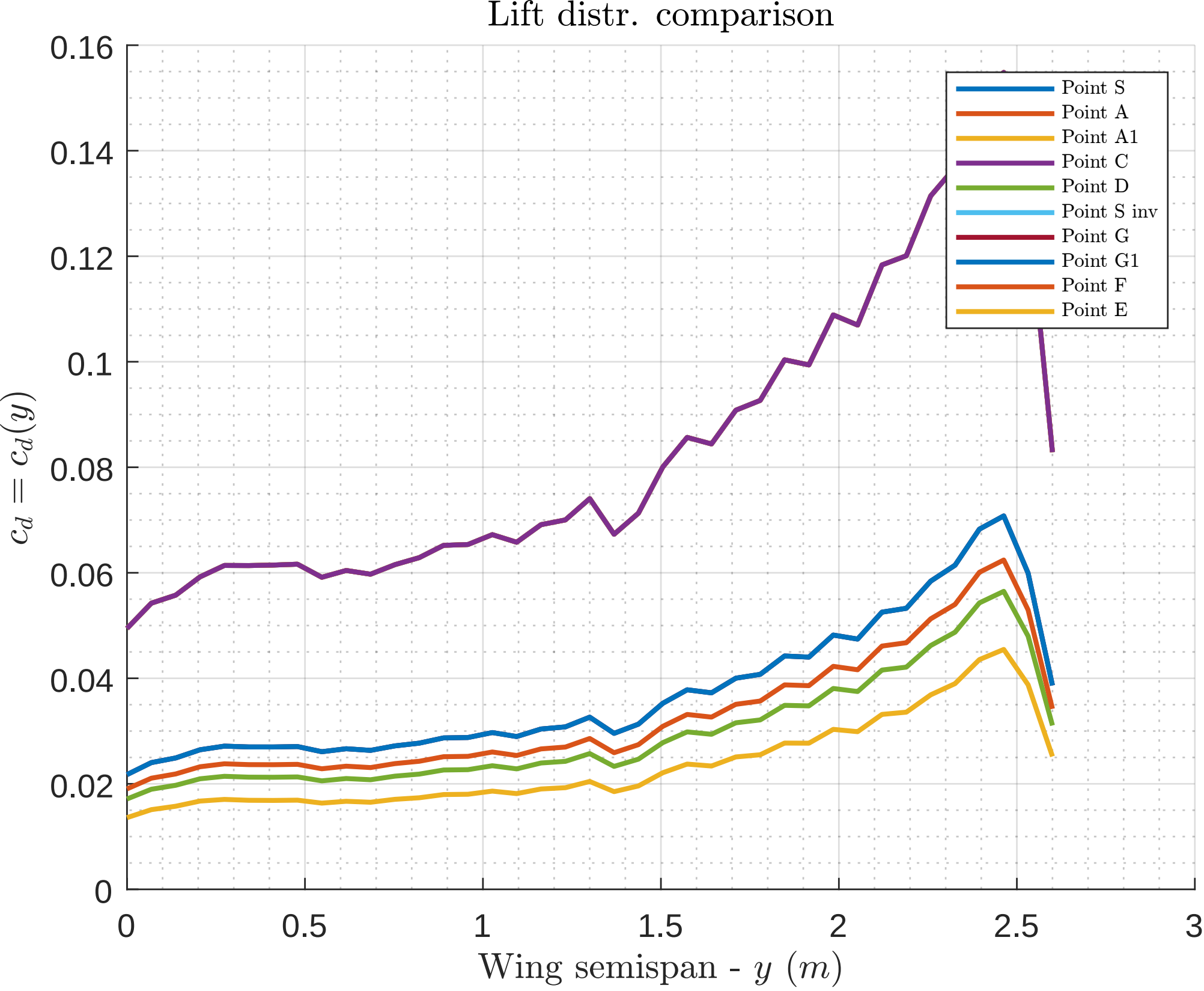
Lift distr. comparison
0.16
0.14
0.12
0.1
0.08
0.06
0.04
0.02
0
$c_d = c_d(y)$
0
0.5
1
1.5
2
2.5
3
Wing semispan - y (m)
Point S
Point A
Point A1
Point C
Point D
Point S inv
Point G
Point G1
Point F
Point E