

A319/A320/A321

TROUBLE SHOOTING MANUAL

HIGHLIGHTS

REVISION NO. 54 May 01/08

Pages which have been revised are outlined below, together with the Highlights of the Revision

CH/SE/SU C
PAGES

REASON FOR CHANGE

EFFECTIVITY

CHAPTER 53

L.E.P. 1- 1 REVISED TO REFLECT THIS REVISION INDICATING
NEW,REVISED, AND/OR DELETED PAGES

A319/A320/A321

TROUBLE SHOOTING MANUAL

CHAPTER 53

FUSELAGE

LIST OF EFFECTIVE PAGES

N, R or D indicates pages which are New, Revised or Deleted respectively
Remove and insert the affected pages and complete the Record of Revisions and
the Record of Temporary Revisions as necessary

CH/SE/SU	C	PAGE	DATE	CH/SE/SU	C	PAGE	DATE	CH/SE/SU	C	PAGE	DATE
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RECORD OF TEMP. REVISION

L.E.P.	R	1-	1	May01/08
T. of C.			1	Aug01/99

53-OBSV		101	May01/97
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53-35-00		201	Feb01/07
53-35-00		202	Feb01/07

TABLE OF CONTENTS

<u>SUBJECT</u>	<u>CH/SE/SU</u>	<u>C</u>	<u>PAGE</u>	<u>EFFECTIVITY</u>
FAULT SYMPTOMS	53-OBSV		101	ALL
FAIRINGS	53-35-00			
FAULT ISOLATION PROCEDURES			201	ALL
Vibration and/or Noise felt in Mid Cabin			201	ALL

A319/A320/A321

TROUBLE SHOOTING MANUAL

FUSELAGE - FAULT SYMPTOMS

	WARNINGS/MALFUNCTIONS	CFDS FAULT MESSAGES				FAULT ISOLATION PROCEDURE
		SOURCE	MESSAGE	ATA	C	
R	VIBRATION - Belly					533500 P 201 T 810 801
R	fairing vibration					

EFF : ALL

SR0S

53-OBSV

Page 101
May 01/97

TASK 53-35-00-810-801

Vibration and/or Noise felt in Mid Cabin

1. Possible Causes

- Aerodynamic Panel-Edge Seals

2. Job Set-up Information

A. Consumable Materials

REFERENCE

DESIGNATION

R Material No. 08-052 USA AMS-T-23397 AND/OR L-T-80B
SELF ADHESIVE ALUMINIUM TAPE (Ref. 20-31-00)

B. Referenced Information

REFERENCE

DESIGNATION

05-50-00-810-801	Identification of the Cause of In-Flight Airframe Vibrations and/or Noises
ASRM 53-35-11	
ASRM 53-45-12	
AMM 53-25-11-000-001	Removal of the Wing-to-Fuselage Fairings (191AT, 191AB, 192AT, 192AB)
AMM 53-25-11-400-001	Installation of the Wing-to-Fuselage Fairings (191AT, 191AB, 192AT, 192AB)
AMM 53-35-11-000-001	Removal of the Wing-to-Fuselage Fairings
AMM 53-35-11-400-001	Installation of the Wing-to-Fuselage Fairings
SIL 51-008	

3. Fault Confirmation

A. Procedure

- (1) Make sure that the identification of the cause of the vibration is correct. (Ref. TASK 05-50-00-810-801).
- (2) If the flight crew have reported vibrations in the center cabin area, do the fault isolation.

EFF : ALL

SR0S

53-35-00

Page 201
Feb 01/07

4. Fault Isolation**A. Do a visual inspection of the Aerodynamic Panel-Edge Seals**

- (1) If the seals are damaged, or are missing, you must replace the seals (Ref. AMM TASK 53-25-11-000-001) and (Ref. AMM TASK 53-25-11-400-001) or (Ref. AMM TASK 53-35-11-000-001) and (Ref. AMM TASK 53-35-11-400-001), and (Ref. SIL 51-008).
 - (2) If you cannot see damage to the seals, but the vibration continues during the subsequent flight:
 - Temporarily install the high speed tape **BONDING AND ADHESIVE COMPOUNDS** (Material No. 08-052) on the panel edges (Ref. ASRM 53-35-11) and (Ref. ASRM 53-45-12).
 - (3) On the panels 191LB and/or 192LB the installation of high speed tape is permitted along the full length of the seal (lip seal between the wing and fairing panels 191LB and/or 192LB).
- NOTE** : Use sufficient high speed tape to apply to the panel seals and panel edges.
- (4) If there is no vibration during the subsequent flight, remove the high speed tape from one panel.
 - (5) Do step (4) again after each subsequent flight, until you find the panel with the vibration.
 - (6) When you find the panel with the vibration, replace the applicable seals (Ref. AMM TASK 53-25-11-000-001) and (Ref. AMM TASK 53-25-11-400-001) or (Ref. AMM TASK 53-35-11-000-001) and (Ref. AMM TASK 53-35-11-400-001), and (Ref. SIL 51-008).

EFF : ALL

SR05

53-35-00

Page 202
Feb 01/07