J58 (JT11D-20) Preliminary Design and Analysis

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**As part of the INME 4707 course offered at the Department of Mechanical Engineering, University of Puerto Rico at Mayaguez, we are required to model and analyze the thermal performance of a J58 Turbojet engine.**

1. **Nomenclature**

*T0* = ambient temperature

*T1* = inlet temperature

*T2*= compressor inlet temperature

*TD* = fourth stage compressor temperature

*T3* = burner inlet temperature

T4 = turbine inlet temperature

T5 = turbine exit temperature

*T6* = afterburner flame-holder temperature

*T8* = nozzle temperature

NEGT = Nominal Exhaust Gas Temperature

*z* = altitude

*M* = Mach Number

= Thrust

*QLHV* = Fuel Lower Heating Value

1. **Introduction**

This project is composed of an analysis of a turbojet engine and its components using the equations derived in class and parameters found in literature. The results obtained were compared to experimental values found in literature to validate the turbojet engine model. The turbojet engine is the basic jet engine [1]. These engines are used extensively in military aircraft due to their supersonic flight capability. They were also used in commercial flight in airplanes such as the BAC Concorde. Turbojet engines are composed of an inlet, a compressor, a combustion chamber, a turbine, and a nozzle. The inlet reduces the velocity of the air mass before entering the engine. The air then enters the compressor, where its pressure is increased. A turbojet engine may have a single compressor or two compressors in series, a low pressure compressor (LPC) and a high pressure compressor (HPC). Then the air enters the combustion chamber where it is mixed with fuel and ignited. This causes the temperature of the mixture to increase. Next, the hot mixture enters the turbine where work is extracted from the fluid to power the compressor and the pressure of air is reduced. As with the compressor, the turbine might be a single one or it may be high pressure turbine (HPT) and a low pressure turbine (LPT) connected in series. The air-fuel mixture finally exits through the nozzle, where it is accelerated and thrust is generated. Some turbojets, specifically those used in military aircraft, may have an afterburner between the turbine and the nozzle that further increases the temperature and velocity of the air mixture. It is used to improve characteristics such as thrust at takeoff, climb, and the combat performance [ref?]. All the engine components play a part in the net thrust achievable by the engine and the efficiency of the engine.

## Problem Statement

To gain a better understanding of turbojet engines it is important to analyze the engine characteristics over a range of condition to fully grasp the capabilities of the engine. To that end, an analytical model will be developed that describes the impact of changes in component characterization on the overall performance of a turbojet engine. This will be done for a range of conditions to survey the design space.

## Background Information

The engine chosen to be analyzed for this project is the Pratt and Whittney J-58JT11D20 jet engine, commonly referred to as just the J58. This is a single spool (one compressor, one turbine) engine with an afterburner, developed from 1958 and produced from 1964 [Smithsonian ref]. It was used as the engine for the various “Blackbird” variants, the A12; YF-12; and SR-71, which were a series of stealth reconnaissance military aircraft used by the CIA and later by NASA as a research aircraft. This engine is famous for being the first with Mach 3 capabilities and broke a number of speed and altitude records [ref?].

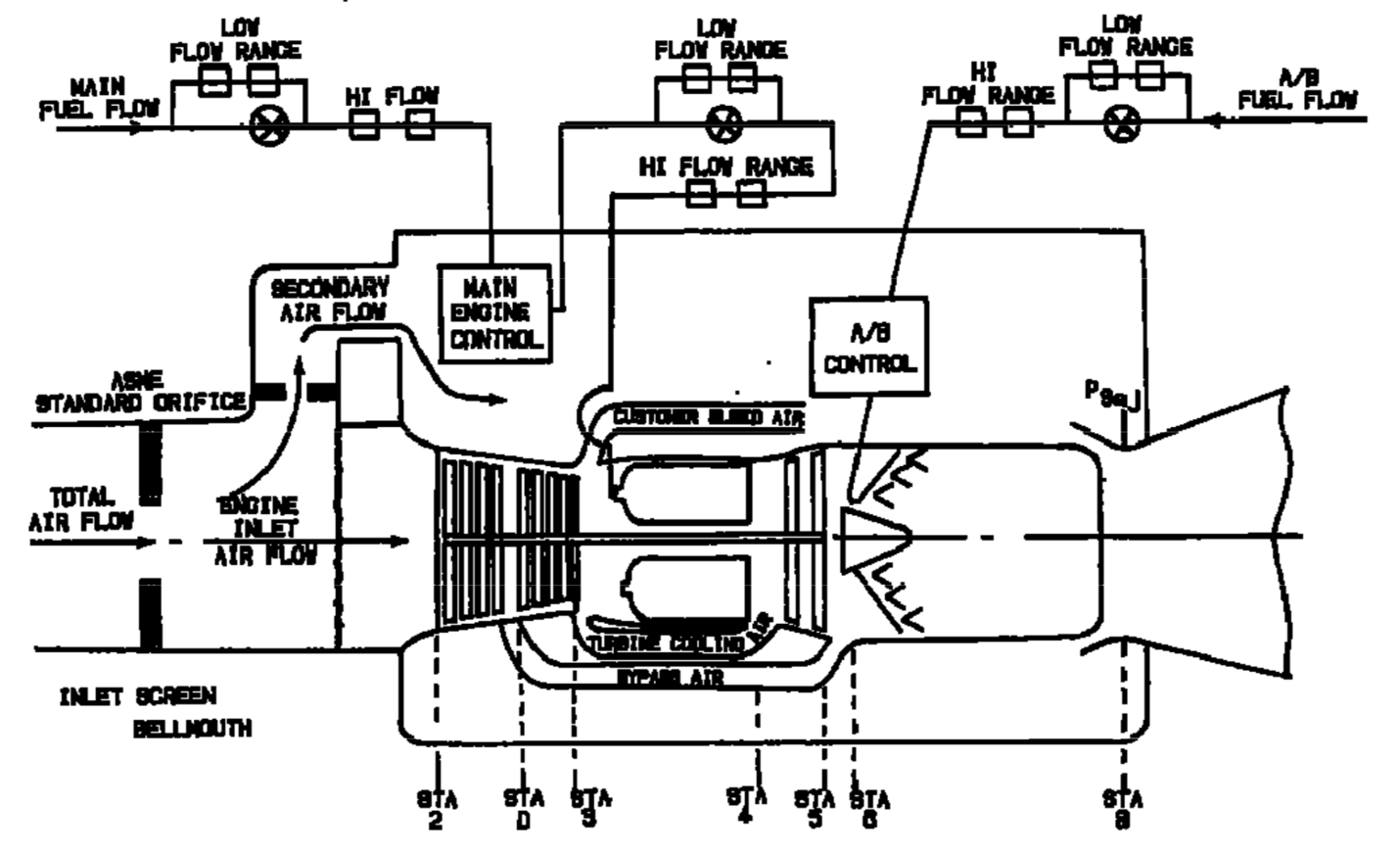


Figure : Standard J11D-20 Station Nomenclature [1]

Table 1: Maximum Operating Temperatures [1] [2]

|  |  |  |
| --- | --- | --- |
| COMPONENT/STAGE | TEMP (ºF) | TEMP (ºC) |
| Inlet T1 | 800+ | 426+ |
| COMPRESSOR Inlet T2 | 800+ | 426+ |
| COMPRESSOR 4th Stage TD | 1050 | 565.56 |
| COMBUSTOR Inlet T3 | 1300 | 704.44 |
| TURBINE Inlet T4 | 2000 | 1093.33 |
| TURBINE Exit T5 | 1450 | 787.78 |
| AB T6 | 3200 | 1760 |
| Exhaust NOZZLE T8 | 1500 | 815.15 |

The JT11D-20 variant of the P&W J58 engine has several components that merit some explanation. For instance, Figure 1 depicts a Bypass Air and Secondary Air Flow; the engine behaved as a traditional afterburning turbojet from subsonic to Mach 2.2, but transitioned to a turboramjet at Mach 2.2 . Above Mach 2.2, 6 valves bypass air from the fourth compressor stage (Station D) to the afterburner thereby combining a turbojet with a compressor assisted ramjet. However, this report will limit the analysis to conditions below Mach 2.2 in order to consider the turbojet nature of the JT11D-20. The secondary airflow depicted in Figure 1 allows “descent at low airflow, low power, without unstarting the inlet.“ [3] (It is also shared with the cowl shock trap bleed as per [3].)

The JT11D-20 was designed for a wide range of operational requirements which included sub- and supersonic flight conditions and a wide range of altitudes. This versatility requires the designed to be evaluated at several conditions which are listed in Table 3. The engine must be capable of performing Buddy Missions, Recon Missions, Long Range Flight Deployments plus the typical Takeoff/Landing conditions. Additionally, the aircraft usually performed high altitude, high Mach flights, but these will not be evaluated due to the Turbo-Ramjet limitation after Mach 2.2. The majority of the flight conditions closely resemble an actual flight condition possibly experienced by an SR-71.

Table 2: Engine Specs

|  |  |  |
| --- | --- | --- |
| SPECIFICATION | VALUE RANGE [EN] | VALUE RANGE [SI] |
| **Altitude** [4] | **25K-90K ft** | **7.62 – 27.43 km** |
| **Speed** [5] | **Mach 0.75 – 3.2** | |
| **Dry TSFC @ Max Thrust** [6] | **0.8 lb/lbf hr** | **81.6 kg/kN hr** |
| **Wet TSFC @ Max Thrust** [6] | **1.9 lb/lbf hr** | **164 kg/kN hr** |
| **Fuel** [7] | **JP-7** | |
| **Fuel Storage** [8] | **80,285 lb** | **36,416 kg** |
| **Fuel Lower Heating Value** [9] | **5.48 kWh/lb** | **43,682 kJ/kg** |
| **Thrust** [7] | **32,500 lbf** | **144,567 N** |
| **Air Volume Flow @ Cruise** [10] | **100K ft3/s** | **2831.68 m3/s** |
| **Compression Ratio < Mach 2.2** [8] | **8.8:1** | |
| **Compressor** [11] | **8-Stage Axial** | |
| **Turbine** [11] | **2-Stage** | |
| **Weight** [11] | **6,500 lb** | **2,948 kg** |
| **Air Mass Flow** [8] | **326-450 lb/s** | **147 – 204 kg/s** |
| **Dry Fuel Mass Flow @ Max** | **5.55 lb/s** | **2.52 kg/s** |
| **Wet Fuel Mass Flow @ Max** | **17.94 lb/s** | **8.14 kg/s** |
| **Dry Fuel to Air Ratio** | **0.012-0.017** | |
| **Wet Fuel to Air Ratio** | **0.0398-0.055** | |

# Methodology: Model Description

Modelling a JT11D-20 requires a non-linear approach; in other terms, the engine requires the coupled equations be solved simultaneously front-to-back and back-to-front in order to better approximate the engine’s actual functioning. For instance, the nominal EGT (T8) is provided by [2] as a function of compressor inlet temperature; therefore, this parameter is fixed once T2 is determined. The inlet design is also a major factor affecting the overall model. Given the supersonic nature of the SR-71 plane, the inlet was designed to minimize the losses incurred by shock waves. The recovery factor is then nonlinear and less than one for a typical flight.

### Ambient

Atmospheric condition will be modelled using [12] based on [13] and [14] with an offset temperature approximating typical aircraft temperatures.

### Inlet

The inlet’s recovery factor will be modelled after the more conservative curve in Figure 8.



Figure 2: Expected Inlet Performance [15]

The air density will be modelled using Equation (1). The speed of sound inside the inlet is then determined as shown in Equation (2). From this change in density and the total pressure recovery, the inlet temperature (T1 or T2 for typical operations) can be determined as seen in Equation (3).

(1)

(2)

(3)

### Nominal EGT

The EGT is given by Figure 9 extracted from [2].

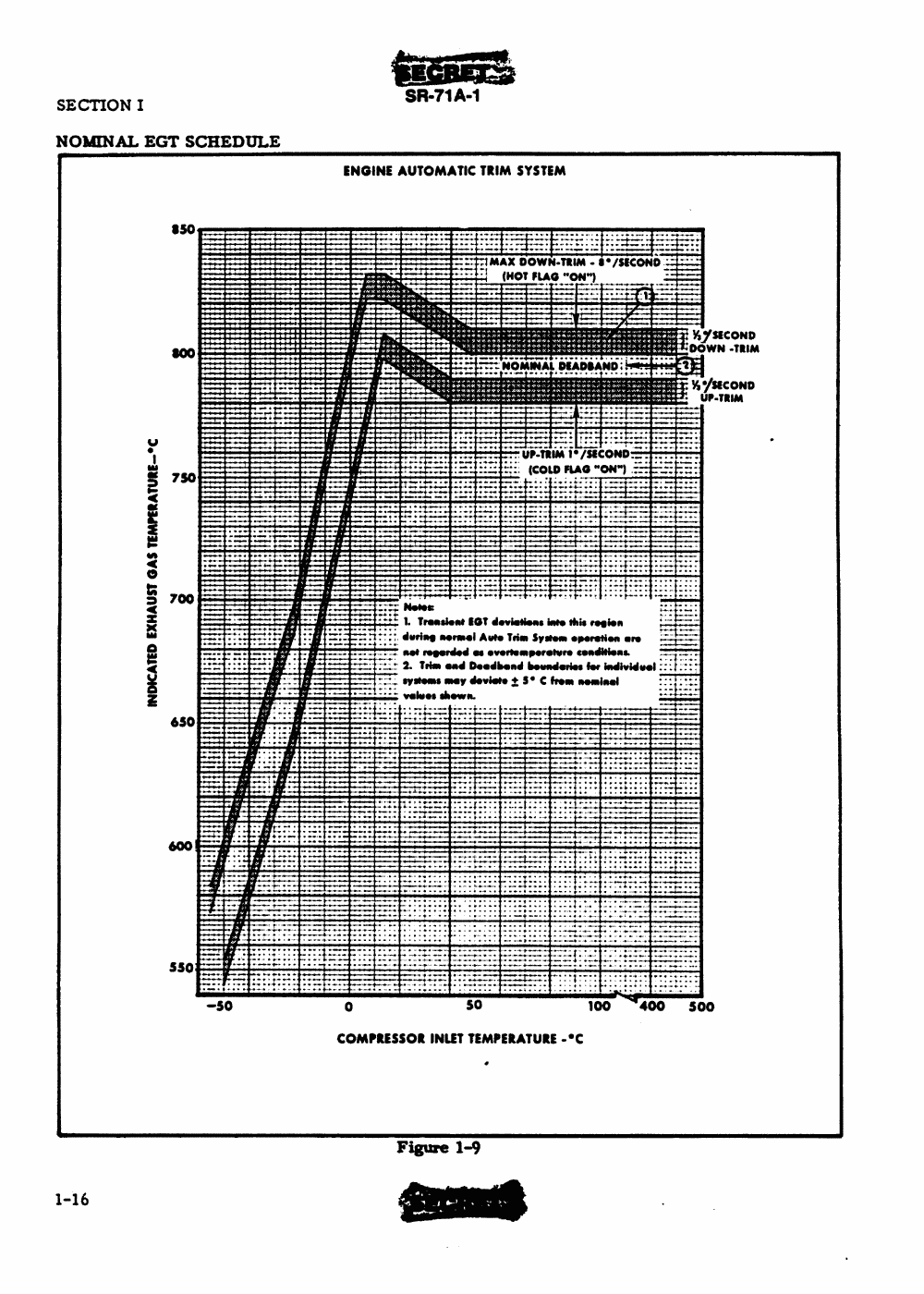


Figure 3: Indicated EGT vs Compressor Inlet Temperature [2]

### Compressor

As per [11], the compression ratio is typically 8.8 and will be assumed constant throughout the model.

### Burner

The JT11D-20’s burner is another source of complexity in the overall design. Albeit the main fuel consumed is JP-7, it is typically mixed with a nitrogen-based additive to promote the ignition of the stable JP-7 [8] [2]. The model assumes JP-7 to be the only fuel present; thereby treating the additive as a neglectable component per unit volume of fuel. Another major assumption presumes the turbine inlet temperature (T4) to remain constant at a maximum value.

The afterburner will be modeled as the burner, however the JP-7 additive assumption is relaxed as the fuel added to the AB is exclusively JP-7.

### Nozzle

The Nozzle’s Area is variable and is a major limiting factor in the numerical modelling still being researched. However, a density-based model was used to determine the volumetric flow at the outlet from which, knowing the speed and mass flow, one could approximate the nozzle’s area.

### Model Validation

The model will be validated at standby with maximum afterburner where a 34000 lbf is expected at 1.9 pounds of JP-7 per hour per pound of thrust generated.

### JT11D-20 Conditions

Table 3: Flight Conditions

|  |  |  |  |
| --- | --- | --- | --- |
| Condition ID | Altitude [ft] | Mach | Afterburner |
| Takeoff [2] | 0 (@ Sea level) | 0.3542 | ON |
| Refueling/Buddy Mission [2] | 25000 | 0.75 | OFF |
| Climbing [2] | 30000 | 1.25 | ON |
| Concorde [16] | 60000 | 2.00 | ON |
| YF12A (03/18/65) [17] | 65000 | 2.2 | ON |
| A12 Max Altitude at Mach 2.2 [18] | 75000 | 2.2 | ON |
| Lake County Airport [18] | 9928 | 0.3545 | ON |
| Lowest Altitude at Mach 1.0 [18] | 15000 | 1.0 | ON |
| MA139-XAA[10] | 40000 | 1.9 | ON |
| French Griffon II [6] | 61000 | 2.1 | ON |
| Constant Climb[5] | 33000 | 0.9 | ON |
| Supersonic Transport flight [19] | 70000 | 2.5 | OFF |

### Implementation

The model’s implementation language is Matlab and the code is being maintained in GitHub for source control facilitation.

# Results and Observations

## Pressure and Temperature Profiles

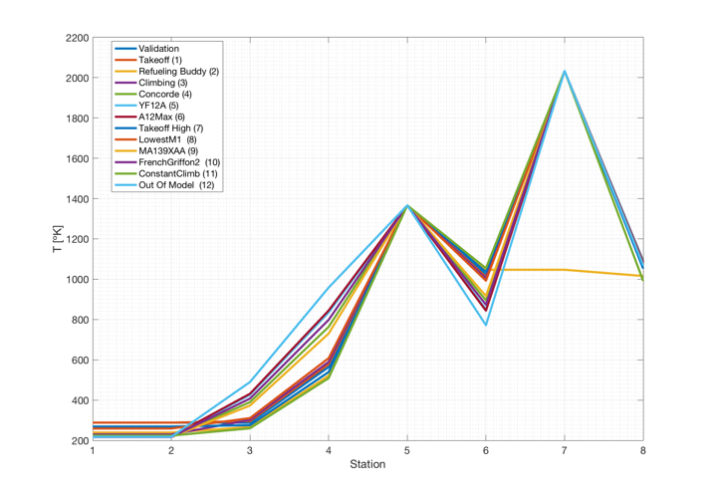


Figure 4: Temperature VS Station ID

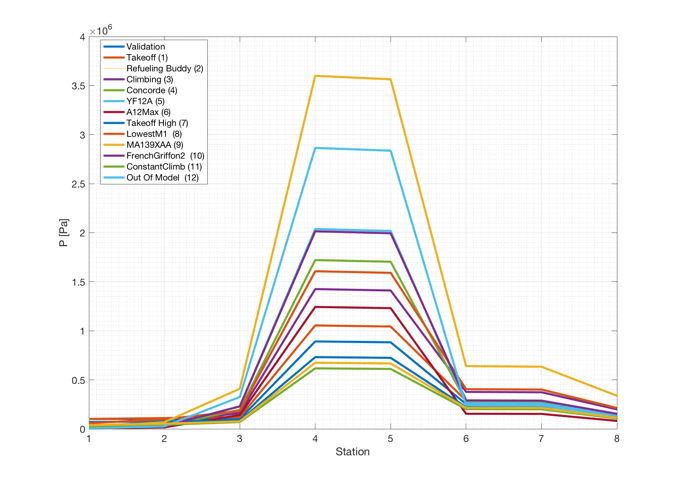


Figure 5: Pressure VS Station ID

Figure 4 shows the temperature profiles for each condition. The graph has the expected shape, exhibiting two peaks in temperature (one at the exit of the combustion chamber and one at the exit of the afterburner) with the highest peak at the afterburner exit for the wet operating conditions and a single peak for the dry operating condition. The pressure graph also has the correct shape, showing a sharp increase in pressure at the compressor, almost constant pressure in the combustion chamber and afterburner, and a sharp decrease in pressure through the turbine.

## Efficiencies

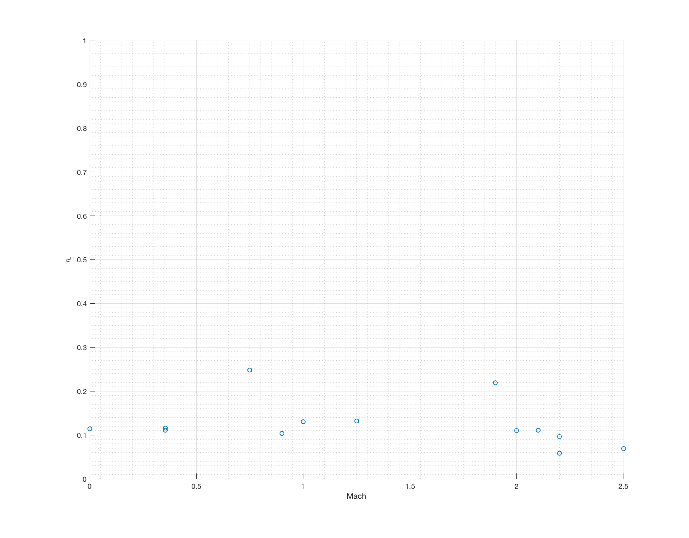


Figure 6: Thermal Efficiency vs Mach

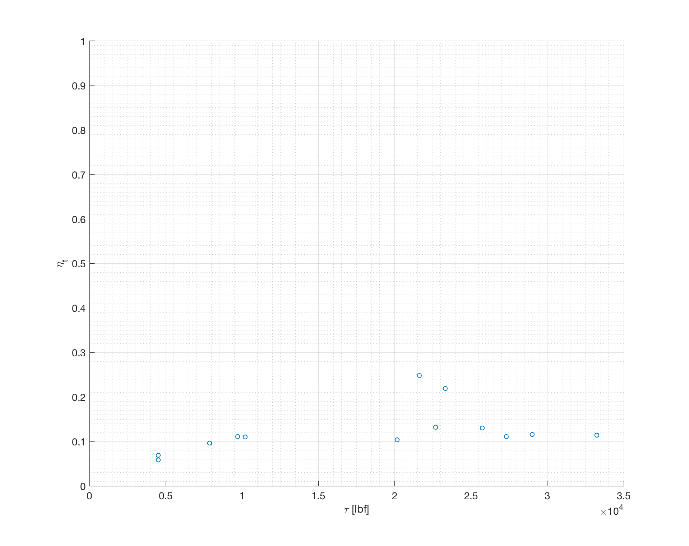


Figure 7: Thermal Efficiency vs Thrust

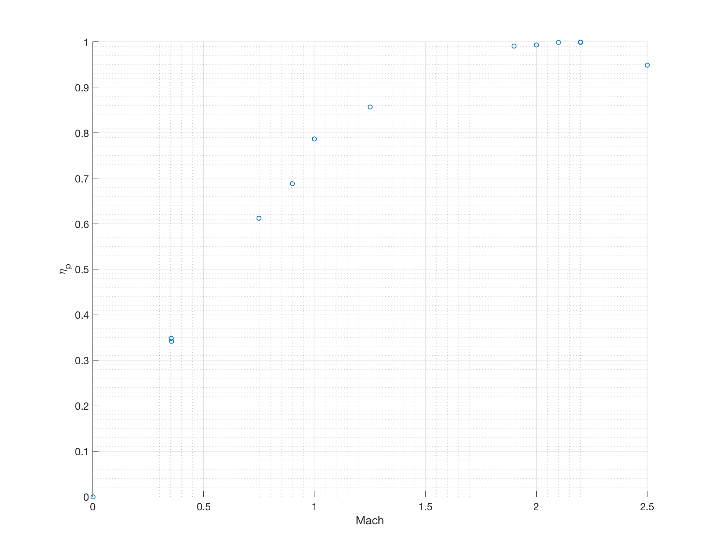


Figure 9: Propulsive Efficiency vs Mach

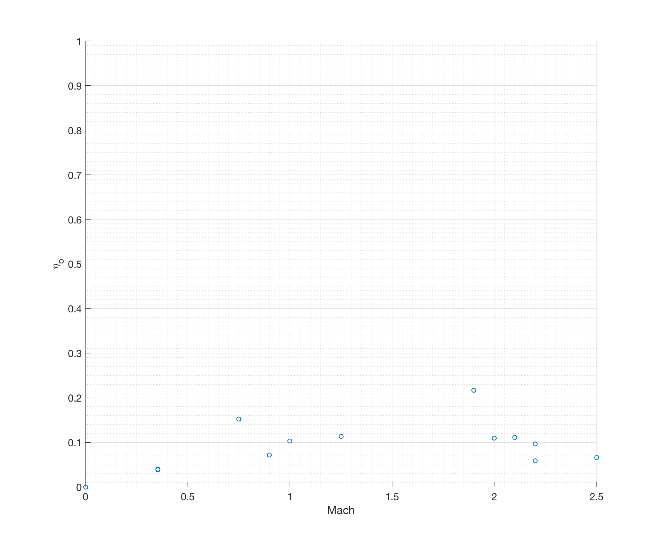


Figure 10: Overall Efficiency vs Mach

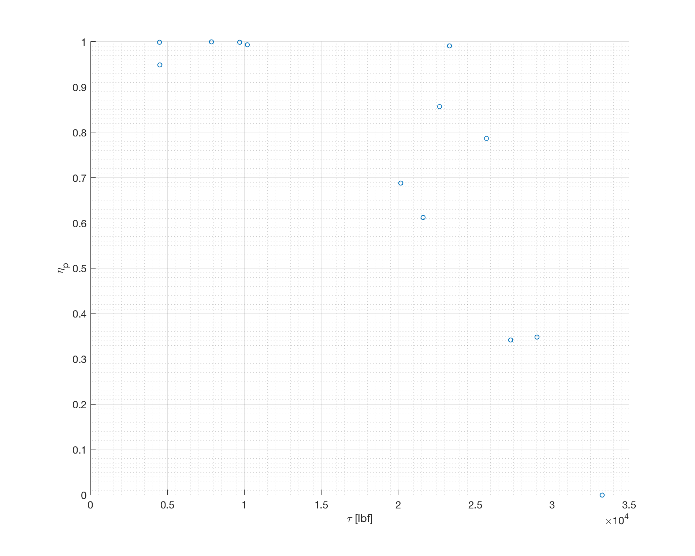


Figure 8: Propulsive Efficiency vs Thrust

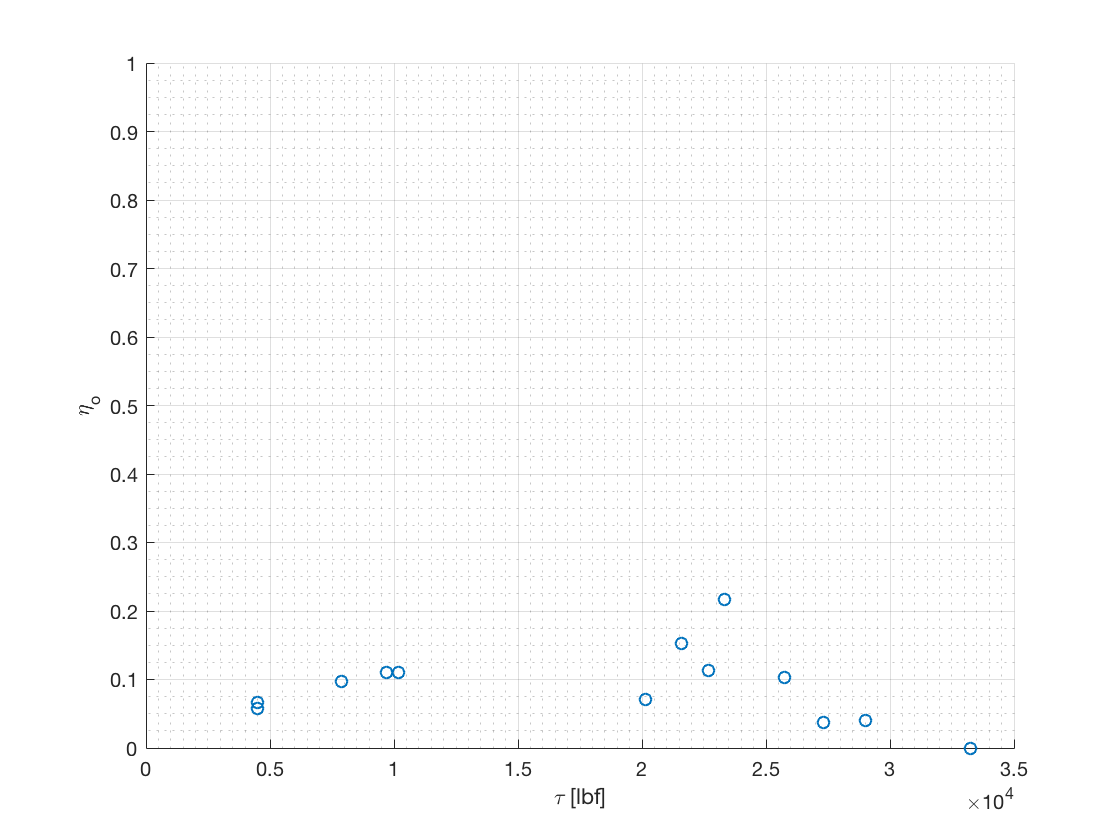


Figure 11: Overall Efficiency vs Thrust

The efficiency graphs are vital in choosing the optimal operating condition for the engine. Given that there is a tradeoff between efficiency and thrust, choosing the condition that maximizes both is complicated. The graphs of efficiency vs thrust are an aid that help to visualize the relationship. From Figure 17 a potential candidate for optimal condition can be observed. A point corresponding to condition 10 is seen to have an higher than the conditions with a lower thrust and significantly higher than those with a higher thrust. The efficiency vs Mach graphs can be used to find the Mach number at which the engine is most efficient, which corresponds to M=1.9 (Condition 10).

## Fuel, Impulse, Range and Specific Thrust

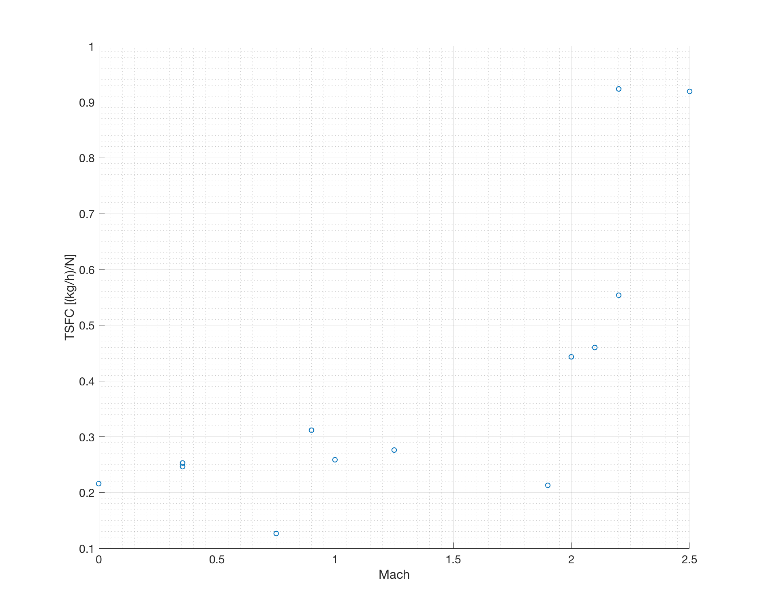


Figure 12: TSFC vs Mach



Figure 13: Fuel to Air Ratio vs Mach

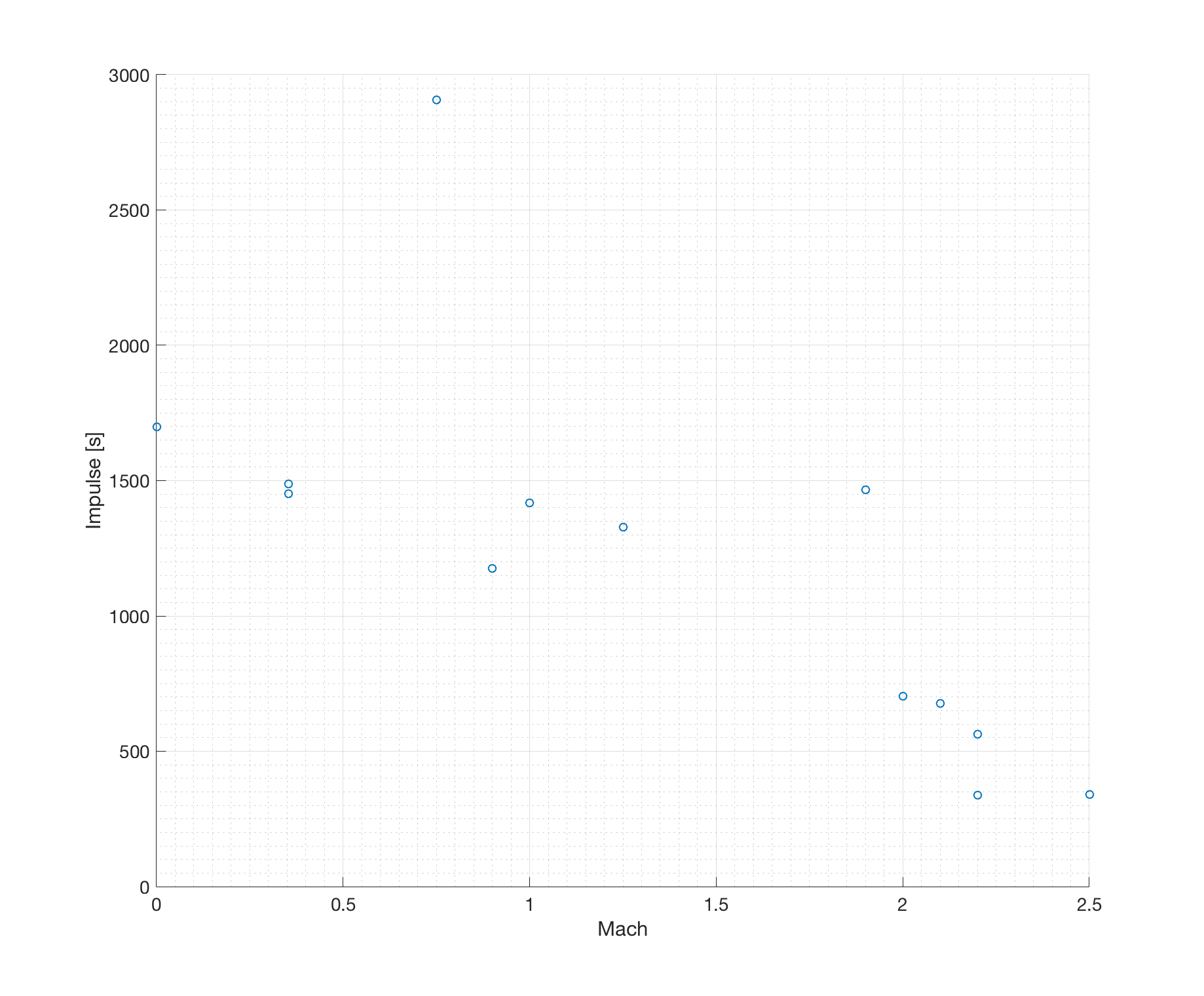


Figure 14: Impulse vs Mach

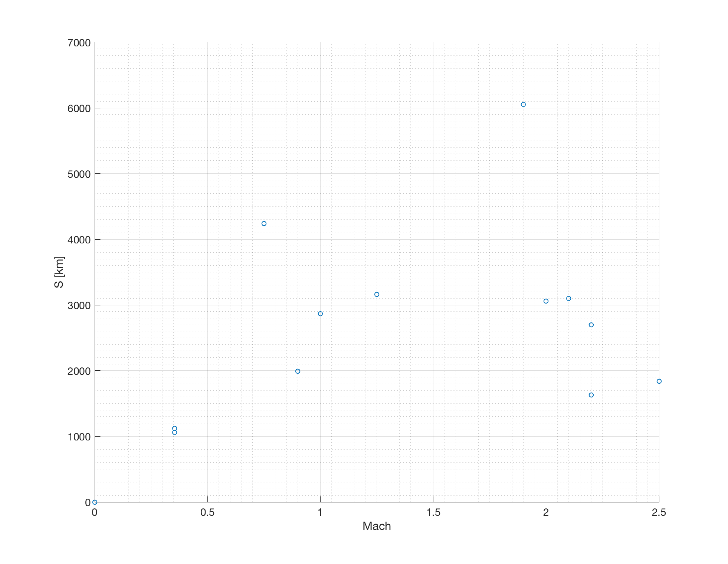


Figure 15: Range vs Mach

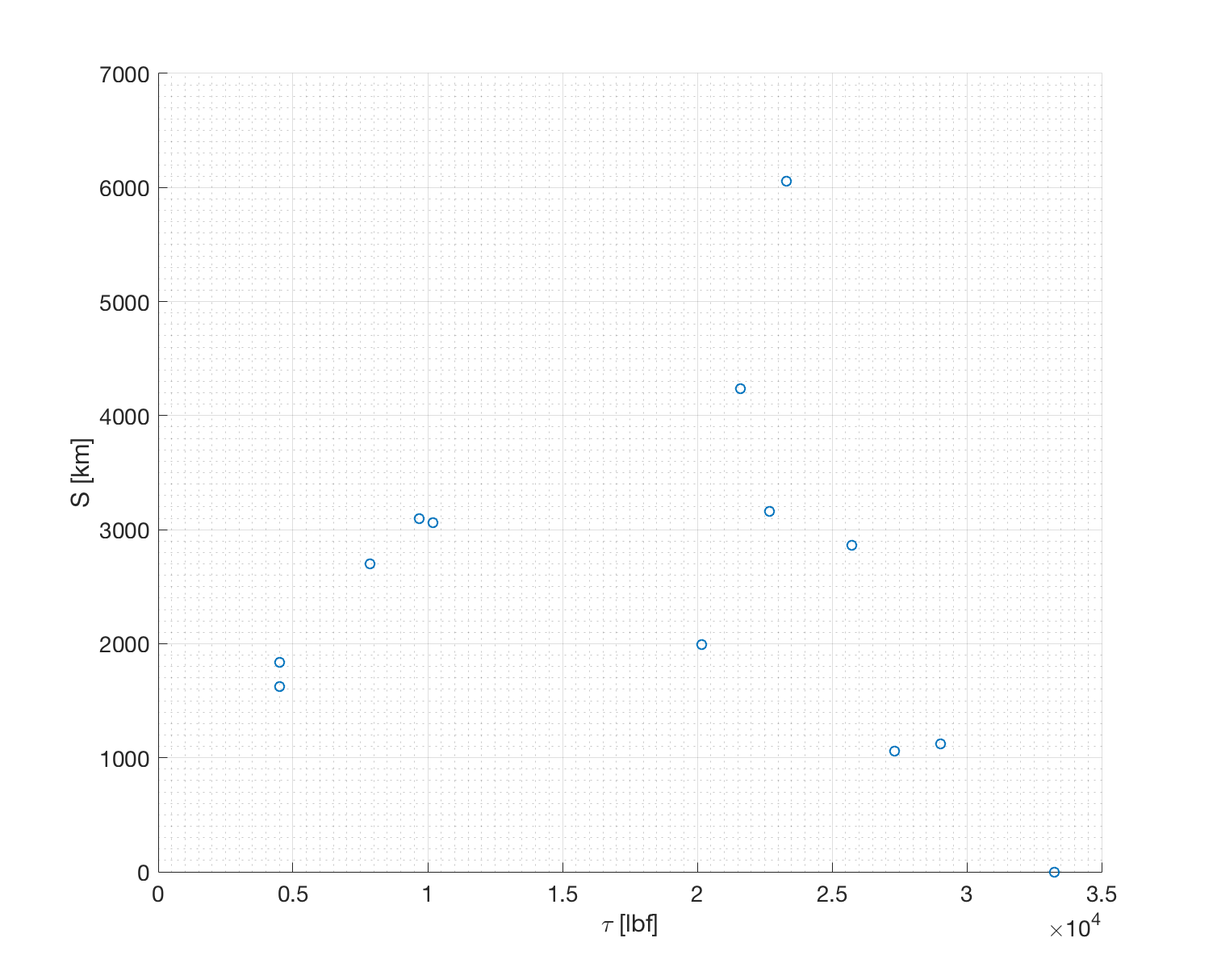


Figure 16: Range vs Thrust

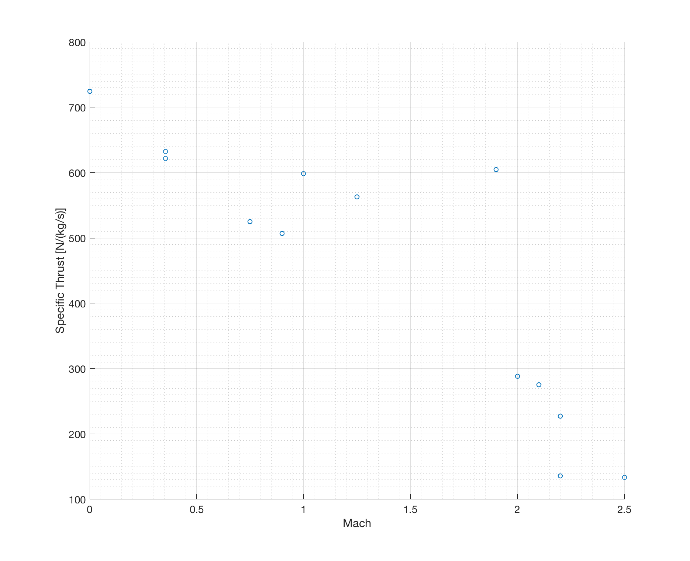


Figure 17: Specific Thrust vs Mach

This section contains additional performance parameters with respect to thrust and Mach. Contrary to part B, where the optimal points corresponded to condition 10 in most cases, here other conditions are more favorable for certain parameters. The TSFC increases significantly for operation conditions with Mach ≥2, and the fuel to air ratio is much higher for wet conditions than for the dry condition. Condition 3 exhibits a large impulse value, condition 10 can operate with the highest range, and the specific thrust is much higher for conditions with Mach <2 than for conditions with Mach ≥2.

## Optimum Determination

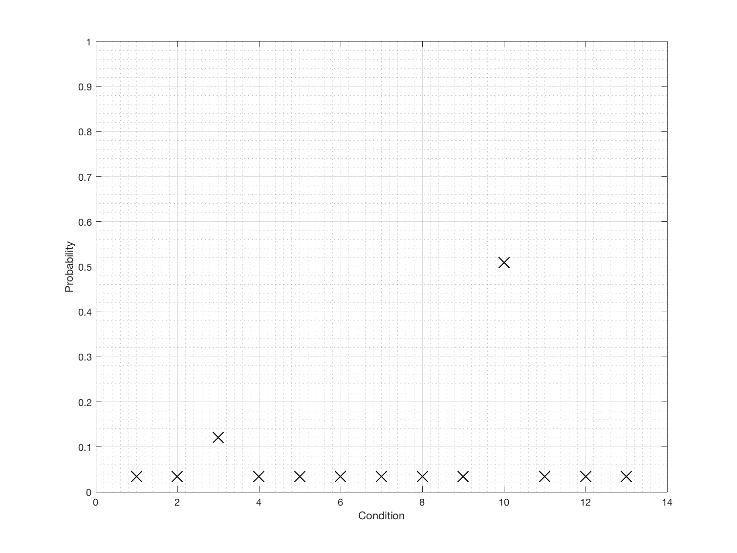


Figure 18: Optimal Selection

A decision matrix was implemented that assigned a weight for each performance parameter to determine the optimal condition. The results of the decision matrix appear graphed above, and it is clear that Condition 10 had the best performance with condition 3 as a far second.

## Summary

|  |  |  |  |  |  |  |  |  |  |  |
| --- | --- | --- | --- | --- | --- | --- | --- | --- | --- | --- |
| Cond. | Mach | Alt. [ft] | AB | [kg/s] | [lbf] |  |  |  | TSFC [kg/N-h] | f |
| 1 | 0 | 0 | ON | 204 | 33241 | 0.114 | 0 | 0 | 2.1187 | 0.0434 |
| 2 | 0.3542 | 0 | ON | 204 | 29018 | 0.115 | 0.348 | 0.040 | 2.4202 | 0.0433 |
| 3 | 0.75 | 25000 | OFF | 183 | 21611 | 0.248 | 0.611 | 0.151 | 1.241 | 0.0184 |
| 4 | 1.25 | 30000 | ON | 179 | 22677 | 0.132 | 0.856 | 0.113 | 2.7182 | 0.0432 |
| 5 | 2.0 | 60000 | ON | 157 | 10196 | 0.110 | 0.992 | 0.109 | 4.3644 | 0.0417 |
| 6 | 2.2 | 65000 | ON | 153 | 7866.8 | 0.096 | 0.999 | 0.096 | 5.4554 | 0.0411 |
| 7 | 2.2 | 75500 | ON | 147 | 4496.8 | 0.058 | 0.998 | 0.058 | 9.1059 | 0.0410 |
| 8 | 0.3545 | 9928 | ON | 195 | 27307 | 0.111 | 0.341 | 0.037 | 2.4841 | 0.0436 |
| 9 | 1.0 | 15000 | ON | 191 | 25733 | 0.130 | 0.786 | 0.102 | 2.5445 | 0.0430 |
| **10** | **1.9** | **40000** | **ON** | **171** | **23318** | **0.218** | **0.990** | **0.216** | **2.0927** | **0.0420** |
| 11 | 2.1 | 61000 | ON | 156 | 9700.1 | 0.111 | 0.998 | 0.110 | 4.5350 | 0.0414 |
| 12 | 0.9 | 33000 | ON | 176 | 20161 | 0.103 | 0.687 | 0.071 | 3.0673 | 0.0439 |
| 13 | 2.5 | 70000 | ON | 150 | 4515.5 | 0.069 | 0.949 | 0.065 | 9.0614 | 0.040 |

# Conclusion

This project had as objective to gain an understanding of the performance capabilities of a selected engine, and to determine the optimal state at which the engine operates from a group of conditions. A computational model was developed to determine the performance parameters of the engine for use in a comparison of the conditions. The conditions were compared graphically across all parameters and the component that exhibited the best values across the conditions was determined to be Condition 10, which corresponds to an airplane flying at Mach 1.9 at an altitude of 40,000 ft. The results of this project highlight the complexity of engine design, since changing small aspects in the operating conditions caused great variations in most performance parameters.

# Appendix

## Group Meetings

INME 4707 Team 1: Group Meeting|Minutes 

## Meeting date | time 04/02/18 | 10:30 | Meeting location Lucchetti

|  |  |  |  |  |  |  |  |  |  |  |  |
| --- | --- | --- | --- | --- | --- | --- | --- | --- | --- | --- | --- |
| |  |  | | --- | --- | | Meeting called by | Christian Lagares | | Type of meeting | Work Division | | Facilitator | Christian Lagares | | Note taker | Christian Lagares | | Timekeeper | Edwin Aponte | | Edwin Aponte  Joel Quijano (Excused from meeting) |

# Agenda topics

## Time allotted | 15 | Agenda topic Source Control | Presenter Christian Lagares

Discuss the importance of Source Control for Code and Document Management and sharing.

Team agreed to a platform

|  |  |  |
| --- | --- | --- |
| Action items | Person responsible | Deadline |
| Create GitHub Account | Joel/Edwin | 04/03/2018 |

## Time allotted | 15 | Agenda topic Code Division | Presenter Christian Lagares

Created basic source tree

Allowed team members to choose desired code sections

|  |  |  |
| --- | --- | --- |
| Action items | Person responsible | Deadline |
| Vote on code section | All | ASAP |



Figure : Work distribution email

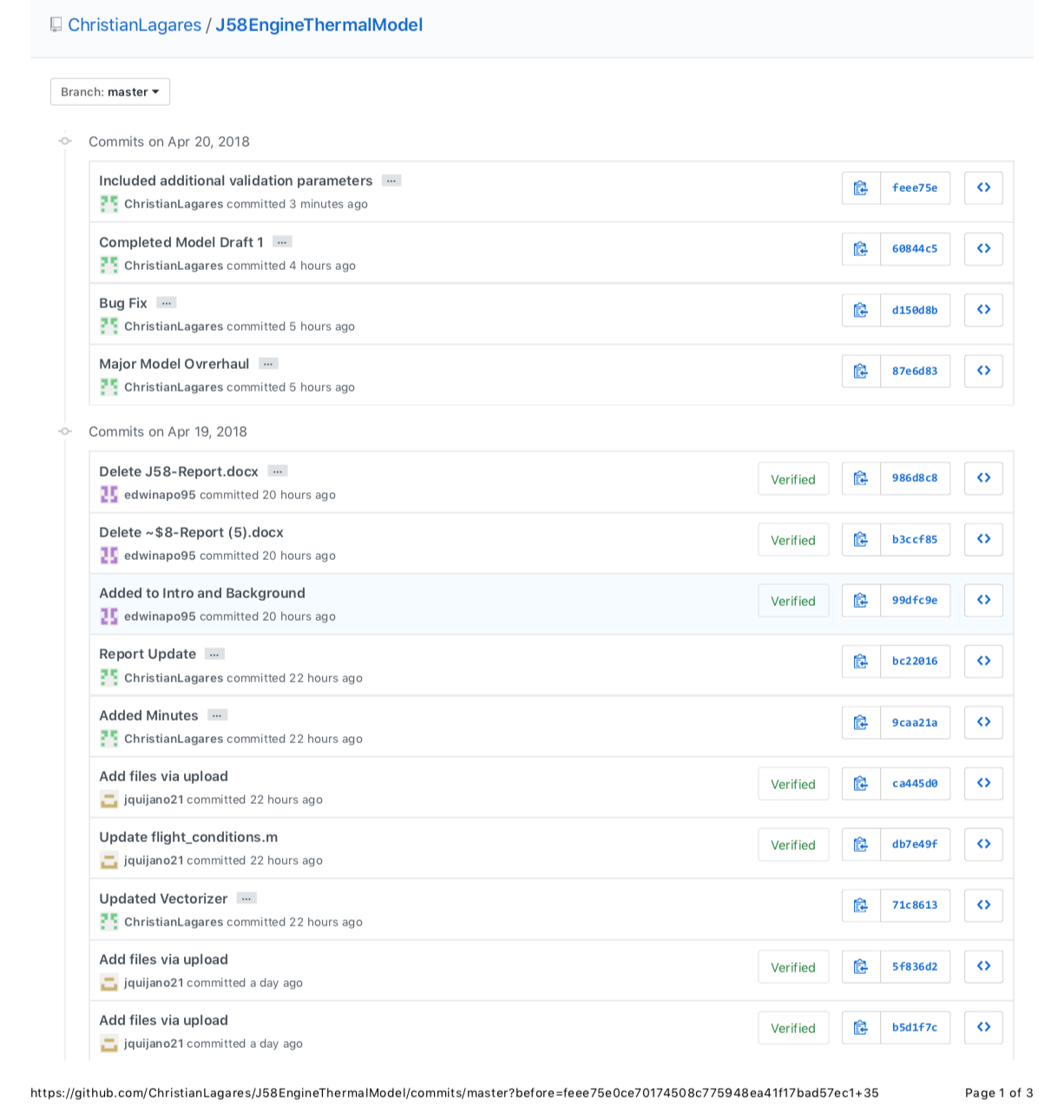


Figure : GitHub Commits; 1 of 5

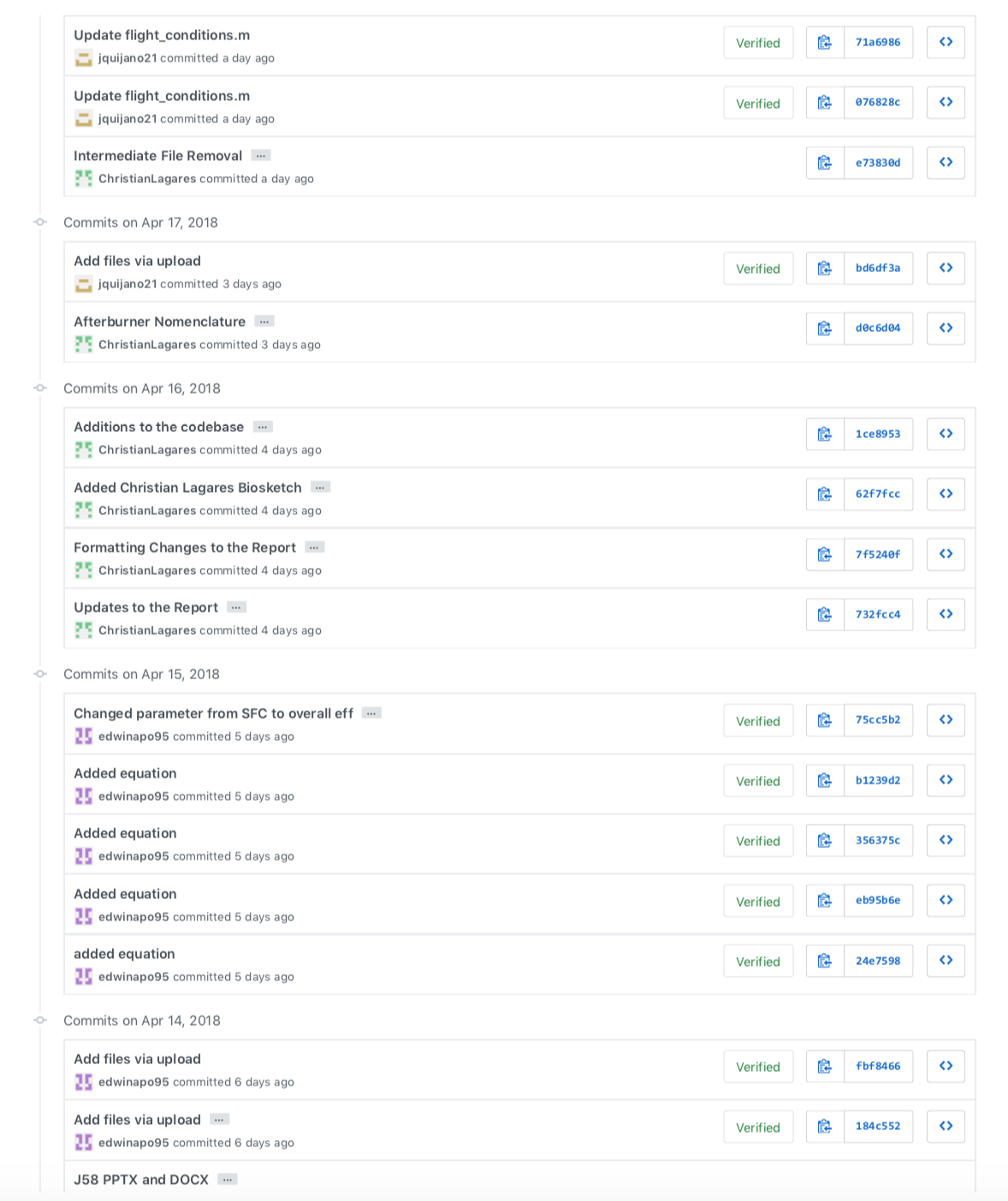


Figure : GitHub Commits; 2 of 5

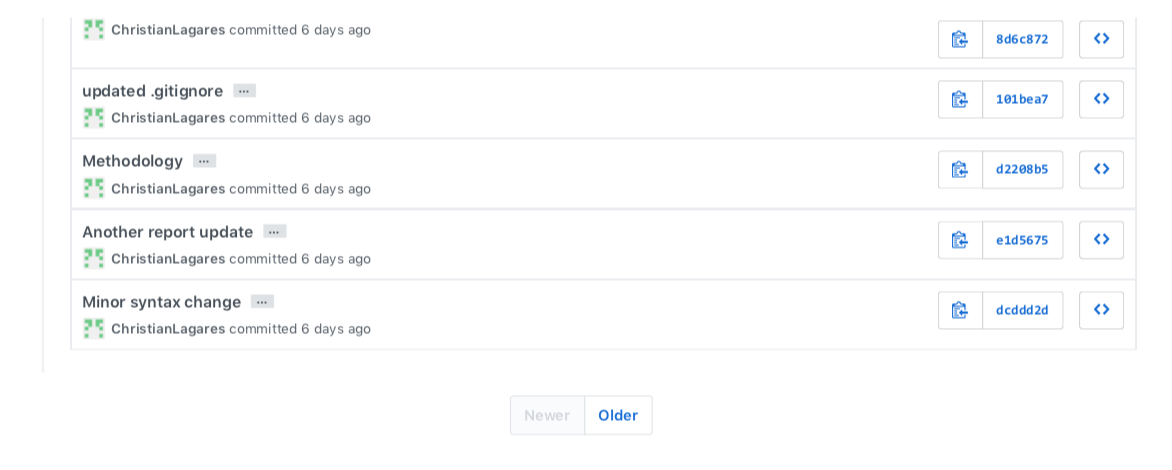


Figure : GitHub Commits; 3 of 5

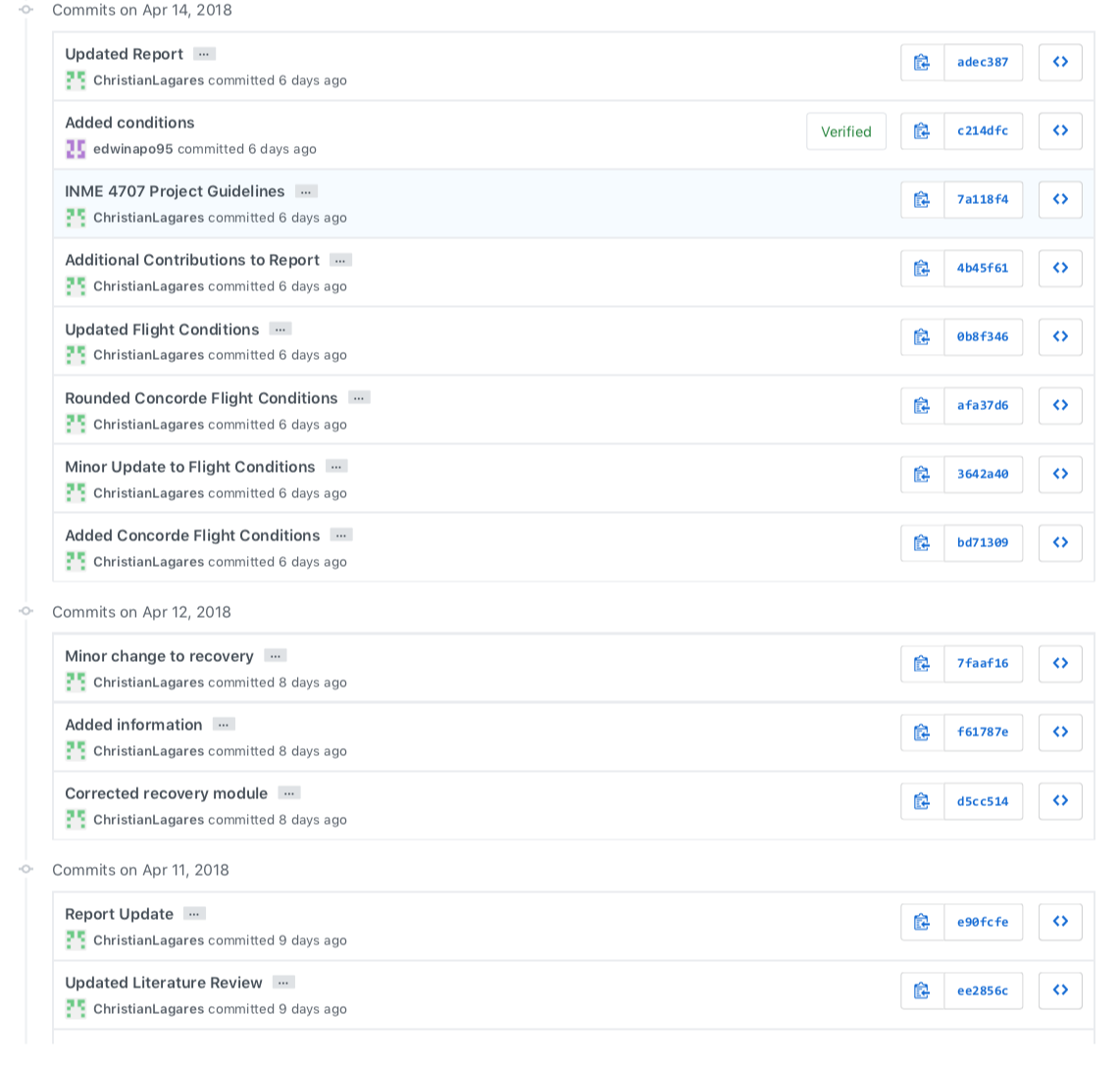


Figure : GitHub Commits; 4 of 5

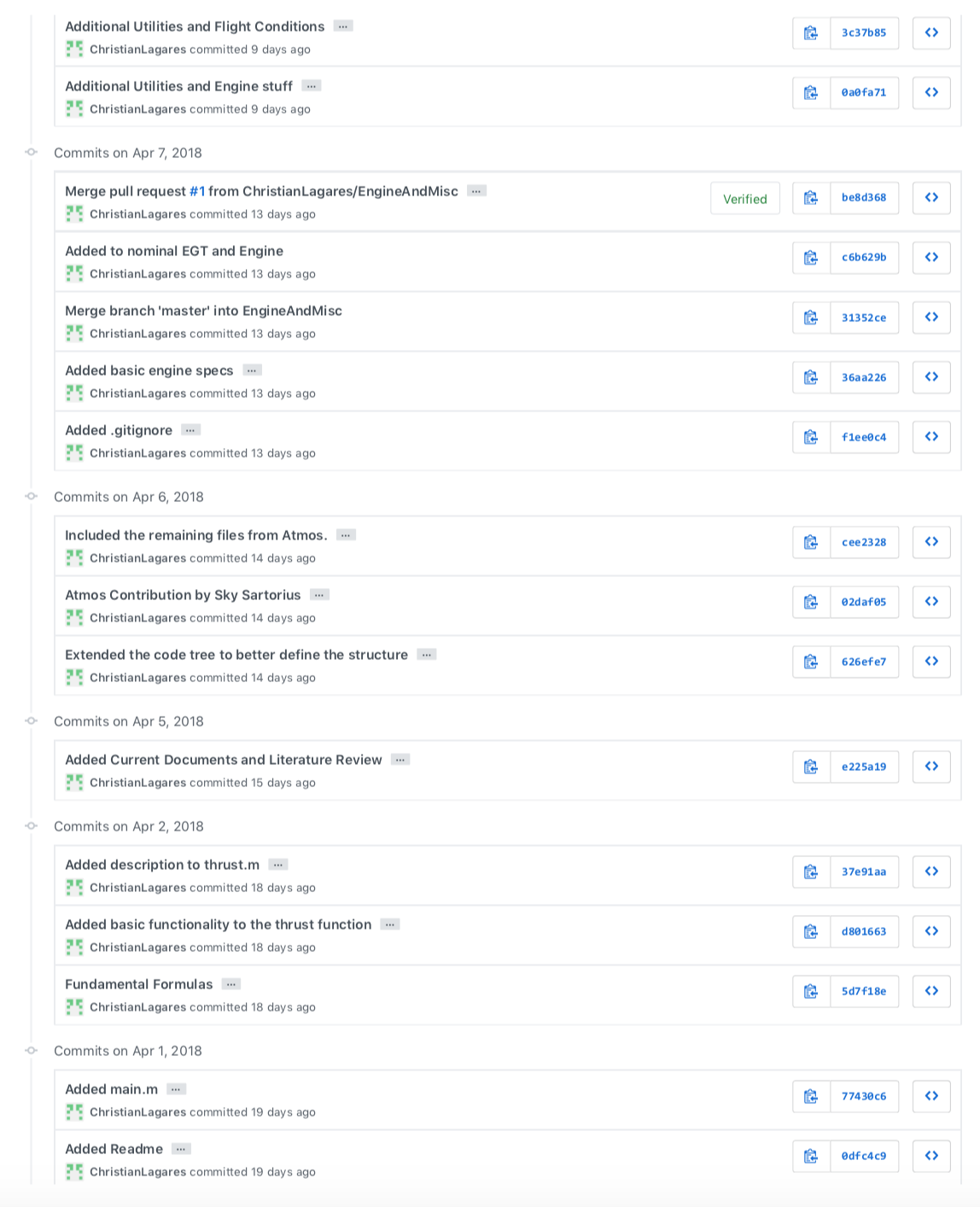


Figure : GitHub Commits; 5 of 5

## Model

The Codebase has been organized as follows:

* EngineModel
  + afterburner.m
  + burner.m
  + compressor.m
  + inlet.m
  + nozzle.m
  + shock\_trap.m
  + turbine.m
* PerfParameters
  + flight\_conditions.m
  + impulse.m
  + overall\_efficiency.m
  + propulsive\_efficiency.m
  + range.m
  + thermal\_efficiency.m
  + thrust.m
  + tsfc.m
* utils
  + atmos
    - atmos.m
    - densityalt.m
    - tropos.m
    - license.txt
  + EngineParameters
    - engine.m
    - recovery.m
  + FlightManualUtilities
    - knots.m
    - nominalEGT.m
* Main.m

#### Codebase - Engine Model – Inlet

%% Inlet

%

% INSERT DOC

%% CODE

function [P\_a, P\_0a, P2, T\_a, T2, mdot\_a, V\_inf] = inlet(altitude, Mach, Toffset)

% inlet - models the SR-71 supersonic inlet

eta\_d = recovery(Mach);

gamma\_c = 1.4;

air\_massflow = struct('ENGLISH',450.\*(exp(-4.27e-06.\*altitude)),...% lb\_s

'SI', 204.\*(exp(-4.34e-06.\*altitude))); % kg/s

mdot\_a = air\_massflow;

altitude = altitude\_converter(altitude, 'ft', 'm');

atmosphere = atmos(altitude, 'units', 'SI', 'structOutput', true);

P\_a = atmosphere.P;

T\_a = atmosphere.T +Toffset;

V\_inf = atmosphere.a .\* Mach;

P\_0a = P\_a + 0.5 .\* atmosphere.rho .\* (Mach .\* atmosphere.a).^2;

P2 = P\_0a.\*((1 + (((gamma\_c - 1)/2).\*(Mach.^2).\*(eta\_d))).^(gamma\_c/(gamma\_c-1)));

T2 = T\_a.\*((1 + (((gamma\_c - 1)/2).\*(Mach.^2))));

end

#### Codebase - Engine Model – Shock Trap

%% Shock Trap

%

% INSERT DOC

%% CODE

function [P2, T2, mdot\_a\_out] = shock\_trap(P2, T2, mdot\_a\_in, Flight\_Mach)

% shock\_trap

mdot\_a\_en\_out = mdot\_a\_in.ENGLISH;

mdot\_a\_si\_out = mdot\_a\_in.SI;

mdot\_a\_en\_in = mdot\_a\_in.ENGLISH;

mdot\_a\_si\_in = mdot\_a\_in.SI;

mdot\_a\_en\_out(Flight\_Mach >= 1.3) = mdot\_a\_en\_in(Flight\_Mach >= 1.3).\*0.85;

mdot\_a\_si\_out(Flight\_Mach >= 1.3) = mdot\_a\_si\_in(Flight\_Mach >= 1.3).\*0.85;

mdot\_a\_out = struct('ENGLISH', mdot\_a\_en\_out,...

'SI', mdot\_a\_si\_out);

end

#### Codebase - Engine Model – Compressor

%% Compressor

%

% INSERT DOC

%% CODE

function [P03, T03, mdot\_a, eta\_c] = compressor(P02, T02, mdot\_a)

% Compressor

eta\_c = 0.9;

gamma\_c = 1.4;

parameters = engine();

compression\_ratio = parameters.COMPRESSION\_RATIO;

P03 = compression\_ratio.\*P02;

T03 = T02.\*(1+(((compression\_ratio.^((gamma\_c-1)/gamma\_c))-1)/eta\_c));

end

#### Codebase - Engine Model – Burner

%% Burner

%

% INSERT DOC

%% CODE

function [P04, T04, fuel2air, mdot\_a, mdot\_e, mdot\_f, JP7LHV] = burner(P03, T03, mdot\_a)

% Docstring

Cph = 1.005; % [kJ/kg ∫K]

Cpc = 1.155; % [kJ/kg ∫K]

JP7LHV = 43682; % [kJ/kg]

T04 = ones(size(T03)).\*(1093.33+273.15); % [∫K] Assumption: Constant @ Nominal

eta\_b = 0.99; % Assumption: From Course Textbook

fuel2air = (Cph.\*T04 - Cpc.\*T03)./(eta\_b.\*JP7LHV - Cph.\*T04);

P04 = P03.\*(1-(1-eta\_b));

mdot\_e = struct('ENGLISH', mdot\_a.ENGLISH.\*(1+fuel2air),...

'SI', mdot\_a.SI.\*(1+fuel2air));

mdot\_f = struct('ENGLISH', mdot\_a.ENGLISH.\*(fuel2air),...

'SI', mdot\_a.SI.\*(fuel2air));

end

#### Codebase - Engine Model – Turbine

%% Turbine

%

% INSERT DOC

%% CODE

function [P05, T05] = turbine(T02, T03, P04, T04, fuel2air)

% Docstring

Cph = 1.005; % [kJ/kg ∫K]

Cpc = 1.155; % [kJ/kg ∫K]

losses = 0.90; % Assumed Shaft Losses

eta\_t = 0.95;

gamma\_h = 1.33;

T05 = T04.\*(1 - ((((Cpc./Cph).\*T02)./(losses.\*(1+fuel2air).\*T04))).\*((T03./T02) - 1));

P05 = P04.\*(1 - ((1./eta\_t).\*(1-(T05./T04)))).^(gamma\_h./(gamma\_h-1));

end

#### Codebase - Engine Model – Afterburner

%% Afterburner

%

% INSERT DOC

%% CODE

function [P06, T06, mdot\_e2, mdot\_f2, fuel2air] = afterburner(P05, T05, AB, mdot\_e1)

% Docstring

Cph2 = 1.155; % [kJ/kg ∫K]

Cph1 = 1.268; % [kJ/kg ∫K]

JP7LHV = 43682; % [kJ/kg]

T06 = ones(size(T05));

T06(AB == 0) = T05(AB == 0);

T06(AB == 1) = (1760+273.15); % [∫K] Assumption: Constant @ Nominal

eta\_b = 0.99; % Assumption: From Course Textbook

fuel2air = (Cph2.\*T06 - Cph1.\*T05)./(eta\_b.\*JP7LHV - Cph2.\*T06);

P06 = P05;

P06(AB == 1) = P05(AB == 1).\*(1-(1-eta\_b));

tmp1 = zeros(size(T05));

tmp1(AB == 1) = mdot\_e1.SI(AB == 1).\*(1+fuel2air(AB == 1));

tmp2 = zeros(size(T05));

tmp2(AB == 1) = mdot\_e1.SI(AB == 1).\*(1+fuel2air(AB == 1));

mdot\_e2 = struct('ENGLISH', tmp1,...

'SI', tmp2);

tmp1 = zeros(size(T05));

tmp1(AB == 1) = mdot\_e1.ENGLISH(AB == 1).\*(fuel2air(AB == 1));

tmp2 = zeros(size(T05));

tmp2(AB == 1) = mdot\_e1.SI(AB == 1).\*(fuel2air(AB == 1));

mdot\_f2 = struct('ENGLISH', tmp1,...

'SI', tmp2);

end

#### Codebase - Engine Model – Nozzle

%% Nozzle

%

% INSERT DOC

%% CODE

function [P8, T8, V8] = nozzle(P06, T06, AB, T02, P\_a)

% Docstring

eta\_n = 0.98;

R = 287;

T8 = nominalEGT(T02 - 273.15)+273.15;

Pc = zeros(size(P06));

P8 = zeros(size(P06));

V8 = zeros(size(P06));

gamma\_h = 1.33;

Cph = 1.155;

% Inoperant AB

Pc(AB == 0) = P06(AB == 0).\*((1 - (1/eta\_n).\*(gamma\_h-1)./(gamma\_h+1)).^(gamma\_h./(gamma\_h-1)));

P8(AB == 0 & Pc >= P\_a) = Pc(AB == 0 & Pc >= P\_a);

P8(AB == 0 & Pc < P\_a) = P\_a(AB == 0 & Pc < P\_a);

V8(AB == 0 & Pc >= P\_a) = sqrt(gamma\_h .\* R .\* T8(AB == 0 & Pc >= P\_a));

V8(AB == 0 & Pc < P\_a) = sqrt(((2.\*gamma\_h.\*eta\_n.\*R.\*...

T06(AB == 0 & Pc < P\_a))./(gamma\_h-1)).\*...

(1-((P\_a(AB == 0 & Pc < P\_a)./...

P06(AB == 0 & Pc < P\_a)).^((gamma\_h-1)./gamma\_h))));

% Operative AB

Pc(AB == 1) = P06(AB == 1).\*((1 - (1/eta\_n).\*(gamma\_h-1)./(gamma\_h+1)).^(gamma\_h./(gamma\_h-1)));

P8(AB == 1 & Pc >= P\_a) = Pc(AB == 1 & Pc >= P\_a);

P8(AB == 1 & Pc < P\_a) = P\_a(AB == 0 & Pc < P\_a);

V8(AB == 1 & Pc >= P\_a) = sqrt(gamma\_h .\* R .\* T8(AB == 1 & Pc >= P\_a));

V8(AB == 1 & Pc < P\_a) = sqrt(2.\*Cph.\*eta\_n.\*T06(AB == 1 & Pc < P\_a).\*...

(1-((P\_a(AB == 1 & Pc < P\_a)./P06(AB == 1 & Pc < P\_a)).^((gamma\_h-1)./gamma\_h))));

end

#### Codebase – Performance Parameters – Flight Conditions

%% Flight Conditions

%

% INSERT DOC

%% CODE

function [conditions] = flight\_conditions()

% Altitude: [ft]

% Mach as per Atmos

% Validation Condition 0

Condition0 = struct('altitude', 0, ...

'Mach', 0.0,...

'Afterburner', 1);

% Takeoff Appendix 2-3

Condition1 = struct('altitude', 0, ...

'Mach', 0.3542,...

'Afterburner', 1);

% Refueling A3-3 & Buddy Mission A4-2

Condition2 = struct('altitude', 25000, ...

'Mach', 0.75,...

'Afterburner', 0);

% Climbing A3-3

Condition3 = struct('altitude', 30000, ...

'Mach', 1.25,...

'Afterburner', 1);

% Concorde Flight Conditions

Condition4 = struct('altitude', 60000, ...

'Mach', 2.00,...

'Afterburner', 1);

% YF12A Record Flight 03/18/65

Condition5 = struct('altitude', 65000, ...

'Mach', 2.2,...

'Afterburner', 1);

% A12 Max. Altitude at M=2.2

Condition6 = struct('altitude', 75500, ...

'Mach', 2.2,...

'Afterburner', 1);

% Takeoff at high altitude airstrip (Lake County Airport)

Condition7 = struct('altitude', 9928, ...

'Mach', 0.3545,...

'Afterburner', 1);

% Lowest operating altitude at M=1.0

Condition8 = struct('altitude', 15000, ...

'Mach', 1.00,...

'Afterburner', 1);

% MA139-XAA

Condition9 = struct('altitude', 40000, ...

'Mach', 1.9,...

'Afterburner', 1);

% French Griffon II

Condition10 = struct('altitude', 61000, ...

'Mach', 2.1,...

'Afterburner', 1);

% Constant climb

Condition11 = struct('altitude', 33000, ...

'Mach', 0.9,...

'Afterburner', 1);

% Supersonic transport flight

Condition12 = struct('altitude', 70000,...

'Mach', 2.5,...

'Afterburner', 1);

conditions = struct('Condition0', Condition0,...

'Condition1', Condition1,...

'Condition2', Condition2,...

'Condition3', Condition3,...

'Condition4', Condition4,...

'Condition5', Condition5,...

'Condition6', Condition6,...

'Condition7', Condition7,...

'Condition8', Condition8,...

'Condition9', Condition9,...

'Condition10', Condition10,...

'Condition11', Condition11,...

'Condition12', Condition12);

end

#### Codebase – Performance Parameters – Condition Vectorizer

%% Internal Condition Vectorization

%

%

%% Code

function [altitude, Mach, Afterburner] = condition\_vectorizer()

% Docstring

FlightConditions = flight\_conditions();

altitude = [FlightConditions.Condition0.altitude,...

FlightConditions.Condition1.altitude,...

FlightConditions.Condition2.altitude,...

FlightConditions.Condition3.altitude,...

FlightConditions.Condition4.altitude,...

FlightConditions.Condition5.altitude,...

FlightConditions.Condition6.altitude,...

FlightConditions.Condition7.altitude,...

FlightConditions.Condition8.altitude,...

FlightConditions.Condition9.altitude,...

FlightConditions.Condition10.altitude,...

FlightConditions.Condition11.altitude,...

FlightConditions.Condition12.altitude];

Mach = [FlightConditions.Condition0.Mach,...

FlightConditions.Condition1.Mach,...

FlightConditions.Condition2.Mach,...

FlightConditions.Condition3.Mach,...

FlightConditions.Condition4.Mach,...

FlightConditions.Condition5.Mach,...

FlightConditions.Condition6.Mach,...

FlightConditions.Condition7.Mach,...

FlightConditions.Condition8.Mach,...

FlightConditions.Condition9.Mach,...

FlightConditions.Condition10.Mach,...

FlightConditions.Condition11.Mach,...

FlightConditions.Condition12.Mach];

Afterburner = [FlightConditions.Condition0.Afterburner,...

FlightConditions.Condition1.Afterburner,...

FlightConditions.Condition2.Afterburner,...

FlightConditions.Condition3.Afterburner,...

FlightConditions.Condition4.Afterburner,...

FlightConditions.Condition5.Afterburner,...

FlightConditions.Condition6.Afterburner,...

FlightConditions.Condition7.Afterburner,...

FlightConditions.Condition8.Afterburner,...

FlightConditions.Condition9.Afterburner,...

FlightConditions.Condition10.Afterburner,...

FlightConditions.Condition11.Afterburner,...

FlightConditions.Condition12.Afterburner];

end

#### Codebase – Performance Parameters – Impulse

%% Impulse

%

% INSERT DOC

%% CODE

function [output] = impulse(air\_massflow,air2fuel,thrust)

% Docstring

output = air\_massflow.\*air2fuel./thrust

end

#### Codebase – Performance Parameters – Propulsive Efficiency

%% Propulsive Efficiency

%

%

%% CODE

function [output] = propulsive\_efficiency(air\_massflow,air2fuel,exhaust\_velocity,flight\_velocity,thrust)

% Docstring

efprop\_num = flight\_velocity.\*thrust;

efprop\_den = efprop\_num + .5.\*air\_massflow.\*(1+air2fuel).\*(exhaust\_velocity-flight\_velocity).^2;

output= efprop\_num./efprop\_den;

end

#### Codebase - Performance Parameters – Thermal Efficiency

%% Thermal Efficiency

%

% INSERT DOC

%% CODE

function [output] = thermal\_efficiency(air\_massflow,air2fuel,...

exhaust\_velocity,flight\_velocity,...

LHV)

% Docstring

efther\_num = (thrust.\*flight\_velocity)+0.5\*air\_massflow.\*...

(1+air2fuel).\*(exhaust\_velocity-flight\_velocity).^2;

efther\_den = air\_massflow.\*air2fuel.\*LHV;

output = efther\_num./efther\_den;

end

#### Codebase - Performance Parameters – Overall Efficiency

%% overall\_effiency

%

% INSERT DOC

%% CODE

function [output] = overall\_effiency(propulsive\_efficiency,...

thermal\_efficiency)

% Docstring

output = propulsive\_efficiency.\*thermal\_efficiency;

end

#### Codebase - Performance Parameters – Overall Efficiency

%% Range

%

% INSERT DOC

%% CODE

function [output] = range()

% Docstring

end

#### Codebase – Performance Parameters – Thrust

%% Thrust

%

% Thrust force is considered an important performance parameter for any

% turbojet engine.

%

%% CODE

function [thrust] = thrust(air\_massflow, air2fuel, exhaust\_velocity,...

flight\_velocity, atmospheric\_pressure, exhaust\_pressure, exhaust\_area)

% thrust computes the thrust force for multiple flight conditions. Inputs

% should be floating point vectors with one entry per flight condition.

pthrust = (exhaust\_pressure - atmospheric\_pressure).\*exhaust\_area;

vthrust = air\_massflow.\*((1+air2fuel).\*exhaust\_velocity + flight\_velocity);

thrust = pthrust + vthrust;

end

#### Codebase – Performance Parameters - TSFC

%% TSFC

%

% INSERT DOC

%% CODE

function [output] = tsfc(air\_massflow,air2fuel,thrust)

% Docstring

output = air\_massflow.\*air2fuel./thrust;

end

#### Codebase – Utils – Atmos – License

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#### Codebase – Utils – Atmos – Atmos

function varargout = atmos(h,varargin)

% ATMOS Find gas properties in the 1976 Standard Atmosphere.

% [rho,a,T,P,nu,z] = ATMOS(h,varargin)

%

% ATMOS by itself gives atmospheric properties at sea level on a standard day.

%

% ATMOS(h) returns the properties of the 1976 Standard Atmosphere at

% geopotential altitude h, where h is a scalar, vector, matrix, or ND array.

%

% The input h can be followed by parameter/value pairs for further control of

% ATMOS. Possible parameters are:

% tOffset - Returns properties when the temperature is tOffset degrees

% above or below standand conditions. h and tOffset must be

% the same size or else one must be a scalar. Default is no

% offset. Note that this is an offset, so when converting

% between Celsius and Fahrenheit, use only the scaling factor

% (dC/dF = dK/dR = 5/9).

% tAbsolute - Similar to tOffest, but an absolute air temperature is

% provided (∞K or ∞R) instead of an offset from the standard

% temperature. Supersedes tOffset if both are provided.

% altType - Specify type of input altitude, either 'geopotential' (h)

% or 'geometric' (z). Default altType = 'geopotential'.

% structOutput - When set, ATMOS produces a single struct output with fields

% rho, a, T, P, nu, and either z or h (whichever complements

% input altType). Default structOutput = false.

% units - String for units of inputs and outpus, either 'SI'

% (default) or 'US'. This is ignored if the provided input h

% is a DimVar, in which case all outputs are also DimVars and

% expected tOffset is either a DimVar or in ∞C/∞K.

% Description: SI: US:

% Input: -------------- ----- -----

% h | z Altitude or height m ft

% tOffset Temp. offset ∞C/∞K ∞F/∞R

% Output: -------------- ----- -----

% rho Density kg/m^3 slug/ft^3

% a Speed of sound m/s ft/s

% T Temperature ∞K ∞R

% P Pressure Pa lbf/ft^2

% nu Kinem. viscosity m^2/s ft^2/s

% z | h Height or altitude m ft

%

% ATMOS returns properties the same size as h and/or tOffset (P does not vary

% with temperature offset and is always the size of h).

%

% Example 1: Find atmospheric properties at every 100 m of geometric height

% for an off-standard atmosphere with temperature offset varying +/- 25∞C

% sinusoidally with a period of 4 km.

% z = 0:100:86000;

% [rho,a,T,P,nu,h] = atmos(z,'tOffset',25\*sin(pi\*z/2000),...

% 'altType','geometric');

% semilogx(rho/atmos,h/1000)

% title('Density variation with sinusoidal off-standard atmosphere')

% xlabel('\sigma'); ylabel('Geopotential altitude (km)')

%

% Example 2: Create tables of atmospheric properties up to 30,000 ft for a

% cold (-20∞C), standard, and hot (+20∞C) day with columns

% [h(ft) z(ft) rho(slug/ft≥) sigma a(ft/s) T(R) P(psf) µ(slug/ft-s) nu(ft≤/s)]

% leveraging n-dimensional array capability.

% [~,h,dT] = meshgrid(0,-5000:1000:30000,[-20 0 20]);

% [rho,a,T,P,nu,z] = atmos(h,'tOffset',dT\*9/5,'units','US');

% t = [h z rho rho/atmos(0,'units','US') a T P nu.\*rho nu];

% format short e

% varNames = {'h' 'z' 'rho' 'sigma' 'a' 'T' 'P' 'mu' 'nu'};

% ColdTable = array2table(t(:,:,1),'VariableNames',varNames)

% StandardTable = array2table(t(:,:,2),'VariableNames',varNames)

% HotTable = array2table(t(:,:,3),'VariableNames',varNames)

%

% Example 3: Use the unit consistency enforced by the DimVar class to find the

% SI dynamic pressure, Mach number, Reynolds number, and stagnation

% temperature of an aircraft flying at flight level FL500 (50000 ft) with

% speed 500 knots and characteristic length of 80 inches.

% V = 500\*u.kts; c = 80\*u.in;

% o = atmos(50\*u.kft,'structOutput',true);

% Dyn\_Press = 1/2\*o.rho\*V^2;

% M = V/o.a;

% Re = V\*c/o.nu;

% T0 = o.T\*(1+(1.4-1)/2\*M^2);

%

% This model is not recommended for use at altitudes above 86 km geometric

% height (84852 m / 278386 ft geopotential) but will attempt to extrapolate

% above 86 km (with a lapse rate of 0∞/km) and below 0.

%

% See also ATMOSISA, ATMOSNONSTD, TROPOS,

% DENSITYALT - http://www.mathworks.com/matlabcentral/fileexchange/39325,

% UNITS - http://www.mathworks.com/matlabcentral/fileexchange/38977.

%

% [rho,a,T,P,nu,z] = ATMOS(h,varargin)

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% www.mathworks.com/matlabcentral/fileexchange/authors/101715

%

% References: ESDU 77022; www.pdas.com/atmos.html

defaultUnits = 'SI'; % Alternate: 'US'

defaultStructOutput = false;

%% Parse inputs:

if nargin == 0

h = 0;

end

if nargin <= 1 && ~nnz(h)

% Quick return of sea level conditions.

rho = 1.2250000;

a = 340.293988026089;

temp = 288.15;

press = 101325;

kvisc = 1.46071857273722e-05;

ZorH = 0;

if isa(h,'DimVar')

rho = rho\*u.kg/(u.m^3);

if nargout == 1

varargout = {rho};

return

end

a = a\*u.m/u.s;

temp = temp\*u.K;

press = press\*u.Pa;

kvisc = kvisc\*u.m^2/u.s;

ZorH = ZorH\*u.m;

end

varargout = {rho,a,temp,press,kvisc,ZorH};

return

end

validateattributes(h,{'DimVar' 'numeric'},{'finite' 'real'});

p = inputParser;

addParameter(p,'tOffset',0,@(x)validateattributes(x,{'DimVar','numeric'},...

{'finite' 'real'}));

addParameter(p,'tAbsolute',[],@(x)validateattributes(x,{'DimVar','numeric'},...

{'finite' 'real' 'positive'}));

addParameter(p,'units',defaultUnits);

addParameter(p,'altType','geopotential');

addParameter(p,'structOutput',defaultStructOutput,...

@(x)validateattributes(x,{'numeric','logical'},{'scalar'}));

parse(p,varargin{:});

tOffset = p.Results.tOffset;

tAbsolute = p.Results.tAbsolute;

convertUnits = strcmpi('US',validatestring(p.Results.units,...

{'US' 'SI'},'atmos','units'));

% Flag if I need to convert to/from SI.

geomFlag = strcmpi('geometric',validatestring(p.Results.altType,...

{'geopotential' 'geometric'},'atmos','altType'));

% Flag specifying z provided as input.

structOutput = p.Results.structOutput;

%% Deal with different input types:

dimVarOut = false;

if isa(h,'DimVar')

h = h/u.m;

dimVarOut = true;

convertUnits = false; % Trumps specified units.

end

if isa(tOffset,'DimVar')

tOffset = tOffset/u.K;

% It is allowed to mix DimVar h\_in and double tOffset (or reverse).

end

if isa(tAbsolute,'DimVar')

tAbsolute = tAbsolute/u.K;

end

if convertUnits

h = h \* 0.3048;

tOffset = tOffset \* 5/9;

tAbsolute = tAbsolute \* 5/9;

end

%% Constants, etc.:

% Lapse rate Base Temp Base Geop. Alt Base Pressure

% Ki (∞C/m) Ti (∞K) Hi (m) P (Pa)

D =[-0.0065 288.15 0 101325 % Troposphere

0 216.65 11000 22632.0400950078 % Tropopause

0.001 216.65 20000 5474.87742428105 % Stratosphere1

0.0028 228.65 32000 868.015776620216 % Stratosphere2

0 270.65 47000 110.90577336731 % Stratopause

-0.0028 270.65 51000 66.9385281211797 % Mesosphere1

-0.002 214.65 71000 3.9563921603966 % Mesosphere2

0 186.94590831019 84852.0458449057 .373377173762337];% Mesopause

% Constants:

R = 287.05287; %N-m/kg-K; value from ESDU 77022

% R = 287.0531; %N-m/kg-K; value used by MATLAB aerospace toolbox ATMOSISA

gamma = 1.4;

g0 = 9.80665; %m/sec^2

RE = 6356766; %Radius of the Earth, m

Bs = 1.458e-6; %N-s/m2 K1/2

S = 110.4; %K

K = D(:,1); %∞K/m

T = D(:,2); %∞K

H = D(:,3); %m

P = D(:,4); %Pa

%% Convert from geometric altitude to geopotental altitude, if necessary.

if geomFlag

hGeop = (RE\*h) ./ (RE + h);

else

hGeop = h;

end

%% Calculate temperature and pressure:

% Pre-allocate.

temp = zeros(size(h));

press = temp;

nSpheres = size(D,1);

for i = 1:nSpheres

% Put inputs into the right altitude bins:

if i == 1 % Extrapolate below first defined atmosphere.

n = hGeop <= H(2);

elseif i == nSpheres % Capture all above top of defined atmosphere.

n = hGeop > H(nSpheres);

else

n = hGeop <= H(i+1) & hGeop > H(i);

end

if K(i) == 0 % No temperature lapse.

temp(n) = T(i);

press(n) = P(i) \* exp(-g0\*(hGeop(n)-H(i))/(T(i)\*R));

else

TonTi = 1 + K(i)\*(hGeop(n) - H(i))/T(i);

temp(n) = TonTi\*T(i);

press(n) = P(i) \* TonTi.^(-g0/(K(i)\*R)); % Undefined for K = 0.

end

end

%% Switch between using standard temp and provided absolute temp.

if isempty(tAbsolute)

% No absolute temperature provided - use tOffset.

temp = temp + tOffset;

else

temp = tAbsolute;

end

%% Populate the rest of the parameters:

rho = press./temp/R;

a = sqrt(gamma \* R \* temp);

kvisc = (Bs \* temp.^1.5 ./ (temp + S)) ./ rho; %m2/s

if geomFlag % Geometric in, ZorH is geopotential altitude (H)

ZorH = hGeop;

else % Geop in, find Z

ZorH = RE\*hGeop./(RE-hGeop);

end

%% Process outputs:

if dimVarOut

rho = rho\*u.kg/(u.m^3);

a = a\*u.m/u.s;

temp = temp\*u.K;

press = press\*u.Pa;

kvisc = kvisc\*u.m^2/u.s;

ZorH = ZorH\*u.m;

elseif convertUnits

rho = rho / 515.3788;

a = a / 0.3048;

temp = temp \* 1.8;

press = press / 47.88026;

kvisc = kvisc / 0.09290304;

ZorH = ZorH / 0.3048;

end

varargout = {rho,a,temp,press,kvisc,ZorH};

if structOutput

if geomFlag

ZorHname = 'h';

else

ZorHname = 'z';

end

names = {'rho' 'a' 'T' 'P' 'nu' ZorHname};

varargout = {cell2struct(varargout,names,2)};

end

end

#### Codebase – Utils – Atmos – Tropos

function [rho,a,temp,press,kvisc]=tropos(h\_in,tOffset)

% TROPOS Stripped-down version of atmos, applicable only to the troposphere

% (covers the vast majority of atmospheric flight), for when computation speed

% is a priority.

%

% [rho,a,T,P,nu] = TROPOS(h)

% [rho,a,T,P,nu] = TROPOS(h,dT)

%

% See also ATMOS.

if nargin < 2

tOffset = 0;

end

if nargin < 1

h\_in = 0;

end

dimVarOut = false;

if isa(h\_in,'DimVar')

h\_in = h\_in/u.m;

dimVarOut = true;

end

if isa(tOffset,'DimVar')

tOffset = tOffset/u.K;

% It is allowed to mix DimVar h\_in and double tOffset (or reverse).

end

% h\_in(h\_in>11000 | h\_in<0) = NaN;

TonTi=1-2.255769564462953e-005\*h\_in;

press=101325\*TonTi.^(5.255879812716677);

temp = TonTi\*288.15 + tOffset;

rho = press./temp/287.05287;

a = sqrt(401.874018 \* temp);

kvisc = (1.458e-6 \* temp.^1.5 ./ (temp + 110.4)) ./ rho;

if dimVarOut

rho = rho\*u.kg/(u.m^3);

a = a\*u.m/u.s;

temp = temp\*u.K;

press = press\*u.Pa;

kvisc = kvisc\*u.m^2/u.s;

end

#### Codebase – Utils – Engine Parameters – Engine

%% Engine Specs

%

% Basic engine parameters are provided in a structure.

%

% engine\_specs:

%

% \* WET\_THRUST

% \* DRY\_THRUST

% \* SFC

% \* AFTERBURNER\_FUEL\_perHour

% \* NO\_AFTERBURNER\_FUEL\_perHour

% \* AFTERBURNER\_FUEL\_perSecond

% \* NO\_AFTERBURNER\_FUEL\_perSecond

%% CODE

function [engine\_specs] = engine()

% Published Specs

WET\_THRUST = 34000; % lbf

DRY\_THRUST = 25000; % lbf

WET\_SFC = 1.9; % lb/lbf\*hr

DRY\_SFC = 0.8; % lb/lbf\*hr

CORE\_AIRFLOW = 450; % lb

COMPRESSION\_RATIO = 8.8;

WET\_FUEL\_hr = WET\_SFC \* WET\_THRUST; % lb/hr

DRY\_FUEL\_hr = DRY\_SFC \* DRY\_THRUST; % lb/hr

WET\_FUEL\_sec = WET\_FUEL\_hr/3600; % lb/sec

DRY\_FUEL\_sec = DRY\_FUEL\_hr/3600; %lb/sec

engine\_specs = struct('WET\_THRUST', WET\_THRUST,...

'DRY\_THRUST', DRY\_THRUST,...

'WET\_FUEL\_perHour', WET\_FUEL\_hr,...

'DRY\_FUEL\_perHour', DRY\_FUEL\_hr,...

'WET\_FUEL\_perSecond', WET\_FUEL\_sec,...

'DRY\_FUEL\_perSecond', DRY\_FUEL\_sec,...

'DRY\_SFC', DRY\_SFC,...

'WET\_SFC', WET\_SFC,...

'CORE\_AIRFLOW', CORE\_AIRFLOW,...

'COMPRESSION\_RATIO', COMPRESSION\_RATIO);

end

#### Codebase – Utils – Engine Parameters – Recovery

%% Recover vs Mach

%% Code

function [ratio] = recovery(Mach)

% Docstring

ratio = 0.004105.\*(Mach.^4)+...

-0.02213.\*(Mach.^3)+...

0.0006246.\*(Mach.^2)+...

0.03737.\*(Mach) + 0.951;

end

#### Codebase – Utils – Flight Manual Utilities – Altitude Converter

%% Altitude Converter

%

%

%% Code

function [length] = altitude\_converter(z, given\_units, required\_units)

% Docstring

given\_units = lower(given\_units);

required\_units = lower(required\_units);

if given\_units == 'ft'

if required\_units == 'm'

length = z .\* 0.3;

elseif required\_units == 'km'

length = (z .\* 0.3)./1000;

end

elseif given\_units == 'm'

if required\_units == 'ft'

length = z .\* 3.281;

elseif required\_units == 'km'

length = z./1000;

end

elseif given\_units == 'km'

z = z .\* 1000;

if required\_units == 'm'

length = z;

else

length = altitude\_converter(z, 'm', required\_units);

end

end

end

#### Codebase – Utils – Flight Manual Utilities – Knots

%% knots

%% Code

function speed = knots(x, units)

% Docstring

if units == 'ft/s'

speed = x .\* 1.69;

elseif units == 'mph'

speed = x .\* 1.15;

elseif units == 'm/s'

speed = x .\* 0.51;

elseif units == 'km/h'

speed = x .\* 1.85;

end

end

#### Codebase – Utils – Flight Manual Utilities – Nominal Exhaust Gas Temperature

%% Nominal Exhaust Gas Temperature

%

%

%% CODE

function egt = nominalEGT(CompressorInletTemperature)

assert(all(CompressorInletTemperature) > -50, 'Off Limits');

egt = zeros(size(CompressorInletTemperature));

egt(CompressorInletTemperature <= 10) = 570 + ((810 - 570)/(10-(-50)))...

\*(CompressorInletTemperature(CompressorInletTemperature <= 10)+50);

egt(CompressorInletTemperature > 10 & CompressorInletTemperature < 50)=...

820 + ((780-810)/(50-10))\*...

(CompressorInletTemperature(CompressorInletTemperature > 10 ...

& CompressorInletTemperature < 50) - 10);

egt(CompressorInletTemperature >= 50) = 795;

end

#### Codebase – Main

%% Main Routine

%

% This file contains the main routine for the J58 Thermal Model.

%

% All other files are required to be in the same directory.

%% Routine Body

%

% Pressures [Pa]

% Temperatures [∫K]

% mdot\_a.ENGLISH [lb/s]

% mdot\_a.SI [kg/s]

% Generating Condition Vectors

[altitude, Mach, AB] = condition\_vectorizer();

% Inlet Model

[P\_a, P0A, P02, T\_a, T02, mdot\_a\_nonbleed, V\_inf] = inlet(altitude, Mach, 0);

% Shock Trap Bleed Model

[P02, T02, mdot\_a] = shock\_trap(P02, T02, mdot\_a\_nonbleed, Mach);

% Compressor Model

[P03, T03, mdot\_a, eta\_c] = compressor(P02, T02, mdot\_a);

% Burner Model

[P04, T04, fuel2air, mdot\_a, mdot\_e1, mdot\_f, LHV] = burner(P03, T03, mdot\_a);

% Turbine Model

[P05, T05] = turbine(T02, T03, P04, T04, fuel2air);

% Afterburner Model

[P06, T06, mdot\_e2, mdot\_f2, ab\_fuel2air] = afterburner(P05, T05, AB, mdot\_e1);

total\_fuel2air = struct('ENGLISH', (mdot\_f2.ENGLISH+mdot\_f.ENGLISH)./mdot\_a.ENGLISH,...

'SI', (mdot\_f2.SI+mdot\_f.SI)./mdot\_a.SI);

%total

% Nozzle Model

[P8, T8, V8] = nozzle(P06, T06, AB, T02, P\_a);

fprintf('V\_inf\tV\_ext\tMach\tProportion\n')

for ii = [1:12]

fprintf('%3.2f\t%3.2f\t%3.2f\t%3.2f\n',V\_inf(ii), V8(ii), Mach(ii), V8(ii)/V\_inf(ii))

end

%% Postprocessing

%

% Nozzle Area

rho\_nozzle = P8./(287.\*T8); % kg/m^3

total\_mdot = struct('ENGLISH', (mdot\_a\_nonbleed.ENGLISH + mdot\_f.ENGLISH...

+ mdot\_f2.ENGLISH),...

'SI', (mdot\_a\_nonbleed.SI + mdot\_f.SI + mdot\_f2.SI));

nozzle\_area = (mdot\_a\_nonbleed.SI + mdot\_f.SI + mdot\_f2.SI)./...

(rho\_nozzle + V8); % m^2

% Thrust

[thrust\_SI] = thrust(mdot\_a.SI, total\_fuel2air.SI, ...

V8,V\_inf, P\_a, P8, nozzle\_area); % Newtons

thrust\_ENGLISH = 0.224808943.\* thrust\_SI; % lbf

% Propulsive Efficiency

prop\_efficiency = (V\_inf.\*thrust\_SI)./((V\_inf.\*thrust\_SI)...

+ 0.5.\*total\_mdot.SI.\*((V8 - V\_inf).^2));

% Thermal Efficiency

thermal\_efficiency = ((thrust\_SI.\*V8) + ...

(0.5.\*mdot\_a.SI.\*(1+total\_fuel2air.SI).\*((V8-V\_inf).^2)))./...

((LHV.\*1000).\*(mdot\_f2.SI+mdot\_f.SI));

% Overall Efficiency

overall\_efficiency = prop\_efficiency .\* thermal\_efficiency;

%% Viz

%

figure('Name','PressureVStation')

plot([P\_a; P0A; P02; P03; P04; P05; P06; P8])

xlabel('Station')

ylabel('P [Pa]')

legend('Validation','Takeoff', 'Refueling\_Buddy', 'Climbing', 'Concorde',...

'YF12A', 'A12Max', 'Takeoff\_High', 'LowestM1',...

'MA139XAA', 'FrenchGriffon2', 'ConstantClimb', 'Out\_Of\_Model')

figure('Name','TemperatureVStation')

plot([T\_a; T\_a; T02; T03; T04; T05; T06; T8])

xlabel('Station')

ylabel('T [∫K]')

legend('Validation','Takeoff', 'Refueling\_Buddy', 'Climbing', 'Concorde',...

'YF12A', 'A12Max', 'Takeoff\_High', 'LowestM1',...

'MA139XAA', 'FrenchGriffon2', 'ConstantClimb', 'Out\_Of\_Model')

figure('Name','ThermalEfficiencyVV\_inf')

scatter(V\_inf', thermal\_efficiency')

xlabel('V\_inf [m/s]')

ylabel('\eta\_t')

figure('Name','PropulsiveEfficiencyVV\_inf')

scatter(V\_inf', prop\_efficiency')

xlabel('V\_inf [m/s]')

ylabel('\eta\_p')

figure('Name','OverallEfficiencyVV\_inf')

scatter(V\_inf', overall\_efficiency')

xlabel('V\_inf [m/s]')

ylabel('\eta\_o')

## Biosketch

Christian Lagares is currently an undergraduate student at the Department of Mechanical Engineering at the University of Puerto Rico at Mayaguez and an Artificial Intelligence/Machine Learning Researcher at SIL Technologies, LLC. His main research interests include Supervised Learning Strategies for Advanced Signal Analysis, Real Time Systems for Simultaneous DAQ/Processing in low power portable devices and AI-Enabled Materials.

Edwin Aponte is currently an undergraduate student at the Department of Mechanical Engineering at the University of Puerto Rico at Mayaguez and working as a research assistant at CAOSE laboratory in UPR Mayaguez. His academic interests include applied mathematics in engineering and aerospace engineering.

Joel Quijano is currently an undergraduate student at the Department of Mechanical Engineering at the University of Puerto Rico at Mayaguez and member of the Solar Engineering Research and Racing Team (SERRT) in said university. His areas of interest are specifically directed to aerodynamics and thermal sciences. Currently active as a Aerodynamics Team leader in SERRT in charge of designing, analyzing and manufacturing of an aero-body using composite materials.

## Additional Results

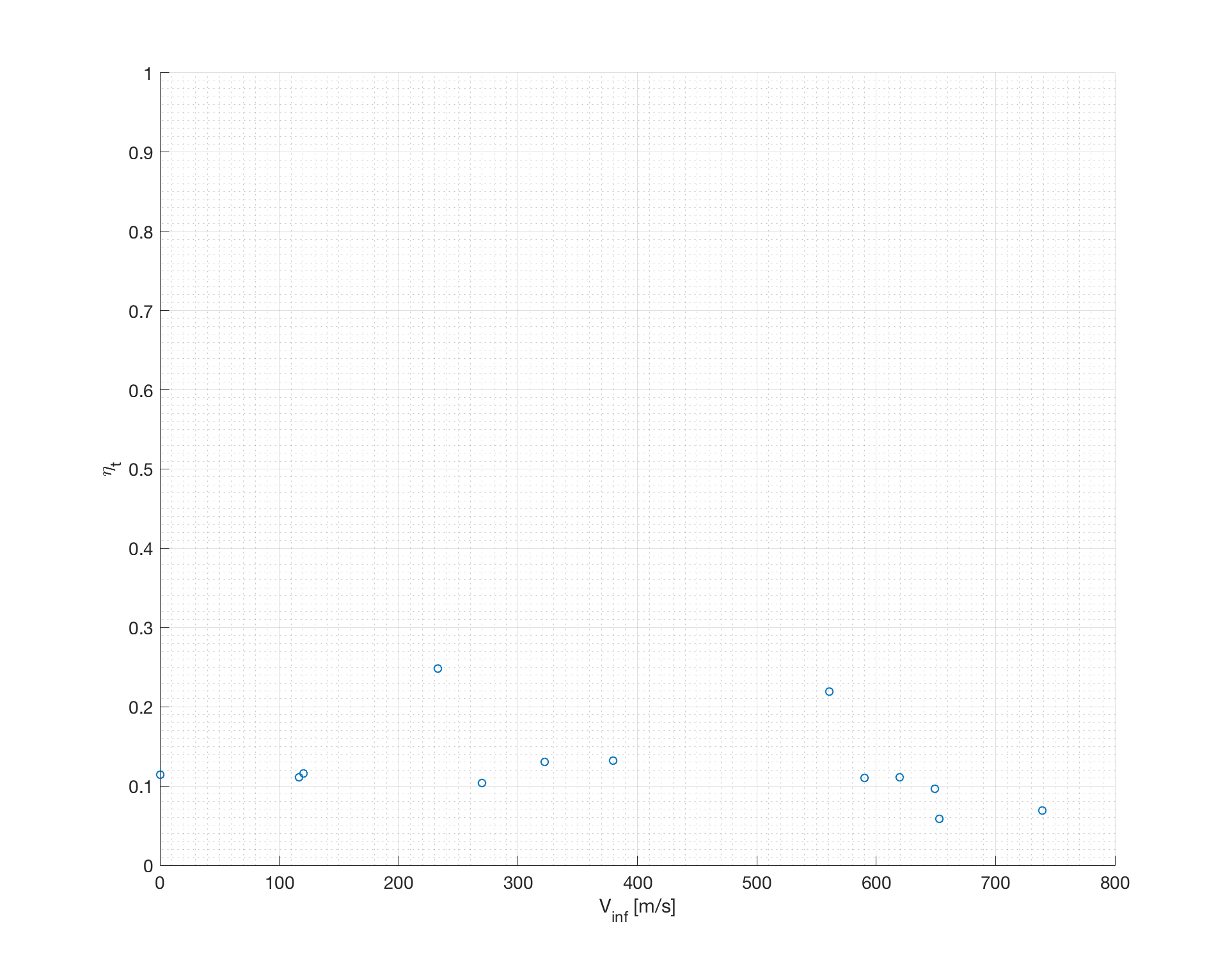


Figure 19: Thermal Efficiency vs Flight Velocity

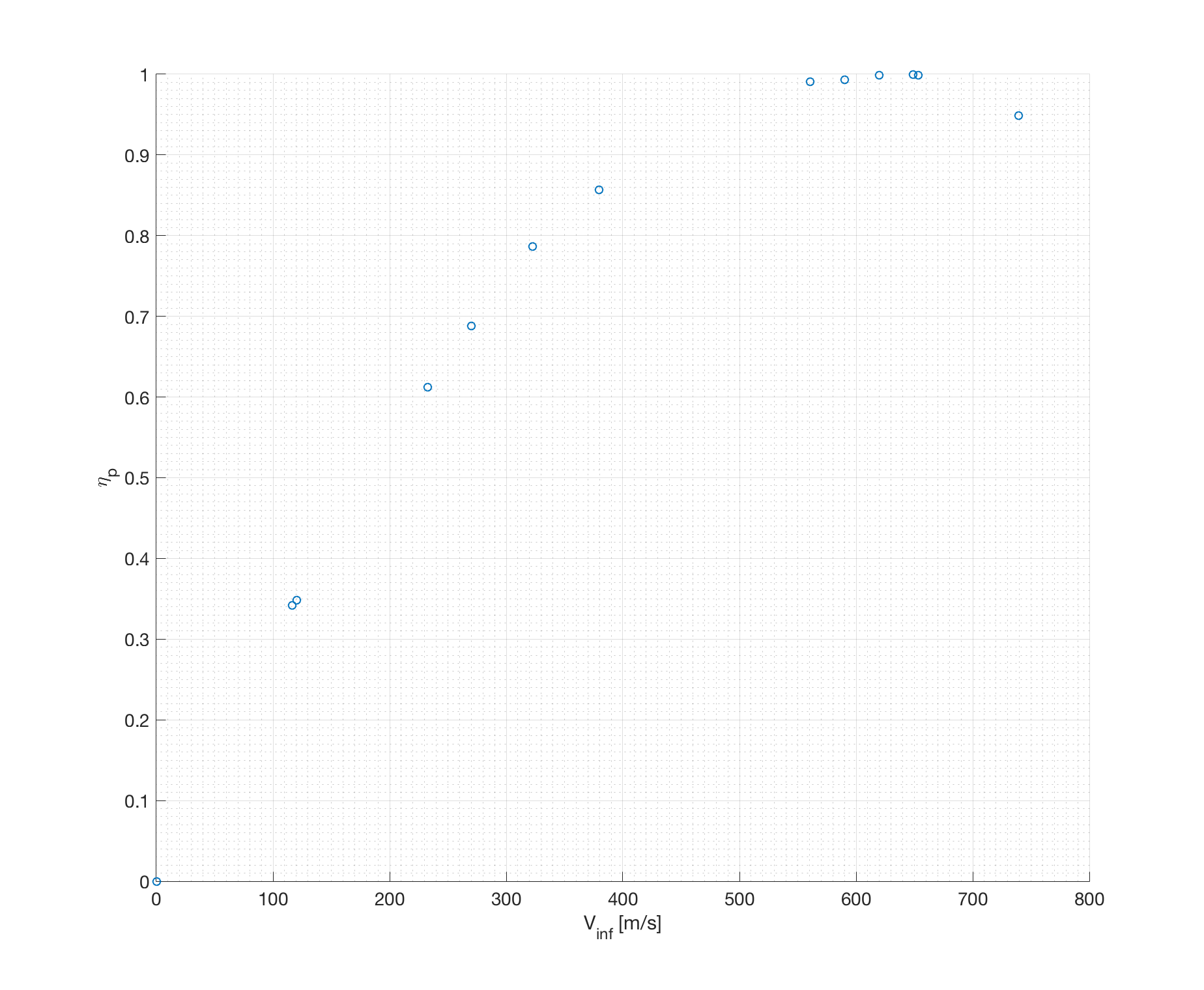


Figure 19: Propulsive Efficiency vs Flight Velocity

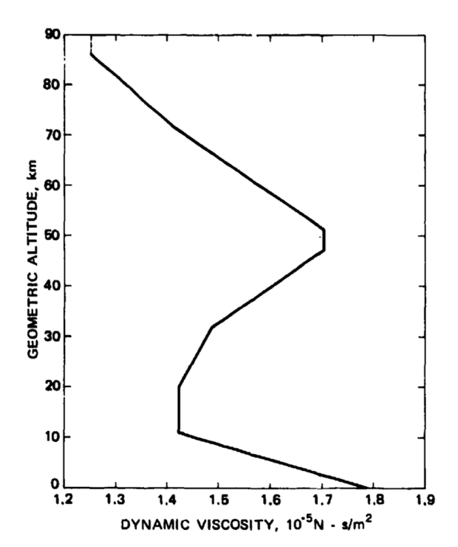


Figure 20: Standard Atmosphere Dynamic Viscosity [14]

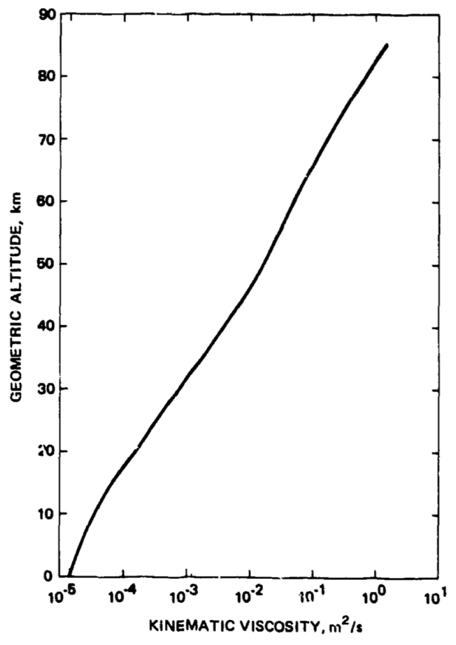


Figure 21: Standard Atmosphere Kinematic Viscosity [14]

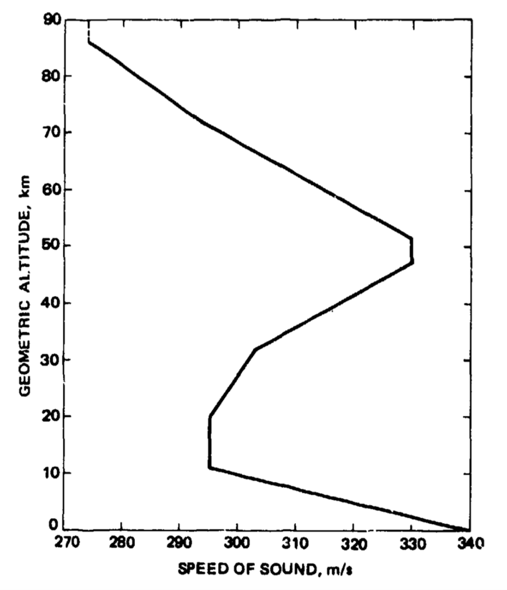


Figure 22: Standard Atmosphere Speed of Sound [14]

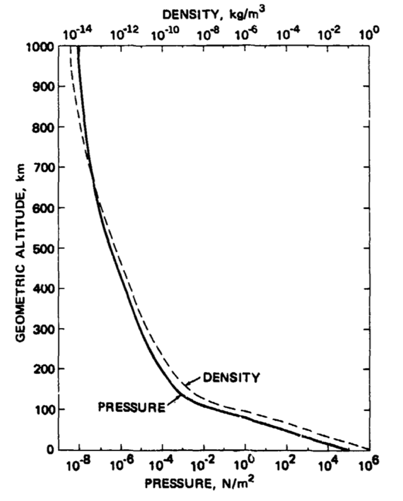


Figure 23: Standard Atmosphere Pressure [14]

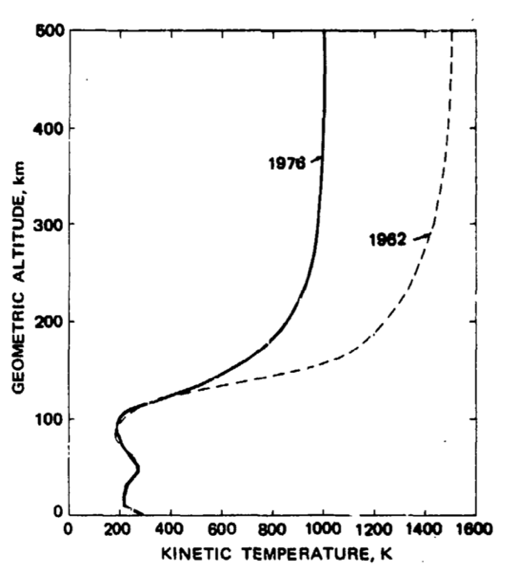


Figure 24: Standard Atmosphere Kinetic Temperature [14]

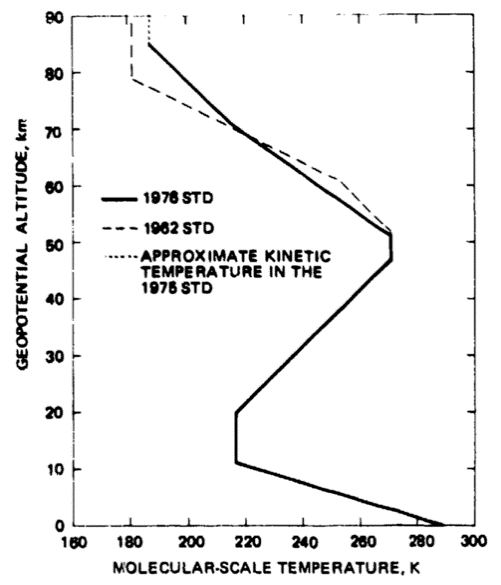


Figure 25: Standard Atmosphere Molecular-Scale Temperature [14]

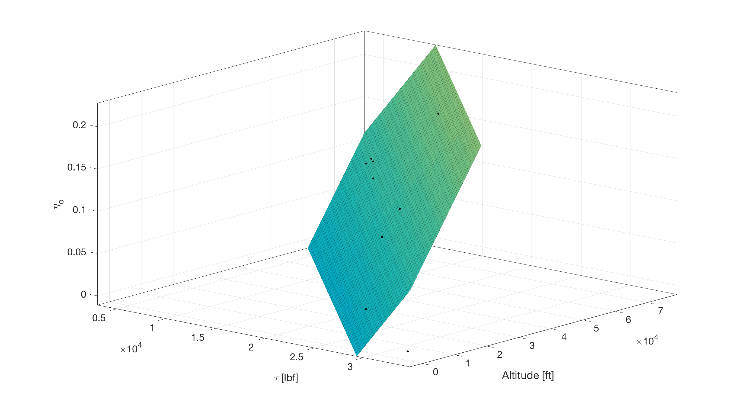


Figure 26:Efficiency vs Thrust 3D Render

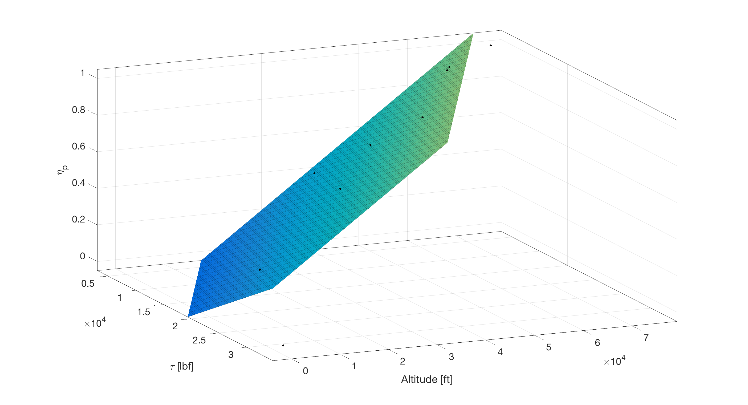


Figure 27:Prop. Eff vs Thrust 3D Render

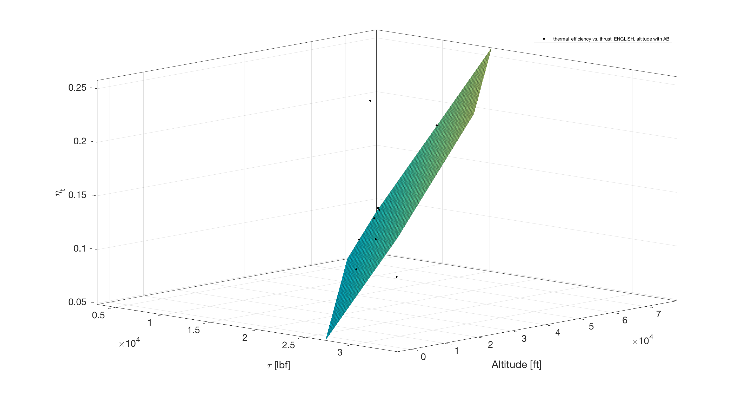


Figure 28:Thermal Eff vs Thrust 3D Render

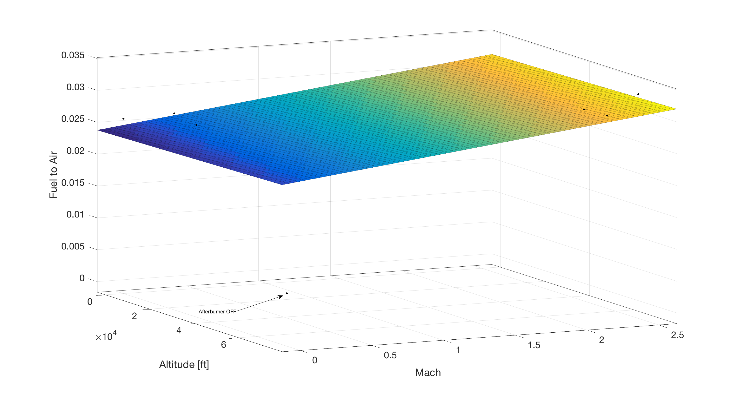


Figure 29:Fuel to Air vs Mach 3D Render

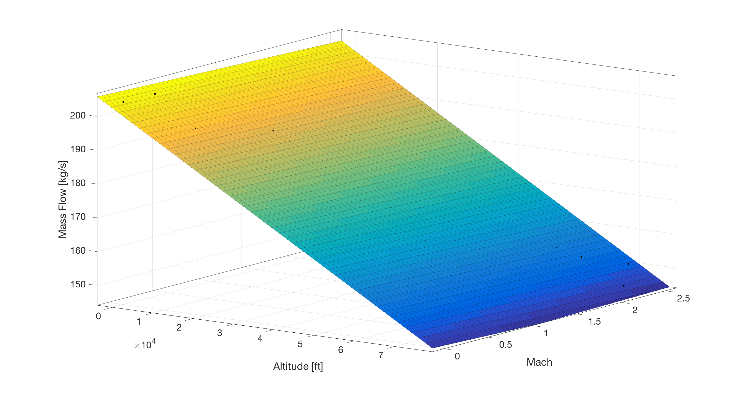


Figure 30:Mass Flow vs Mach 3D Render

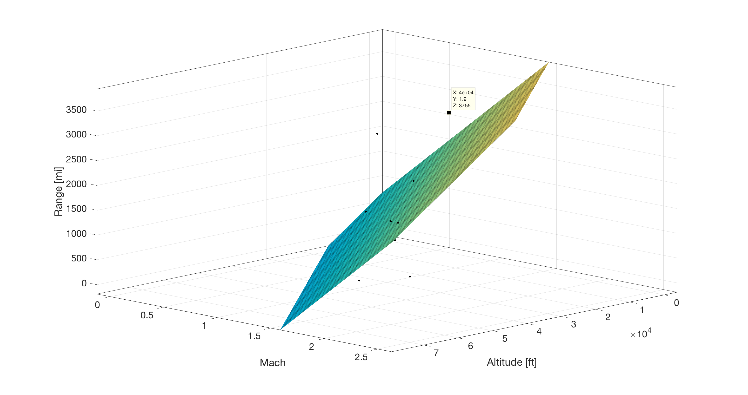


Figure 31:Range vs Mach 3D Render

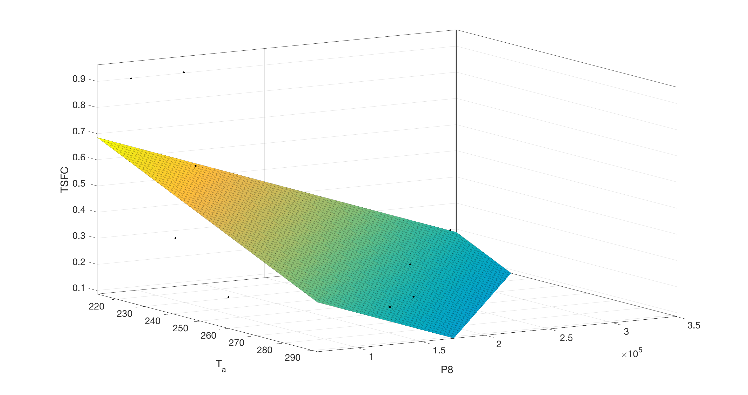


Figure 32:TSFC vs T atm. 3D Render

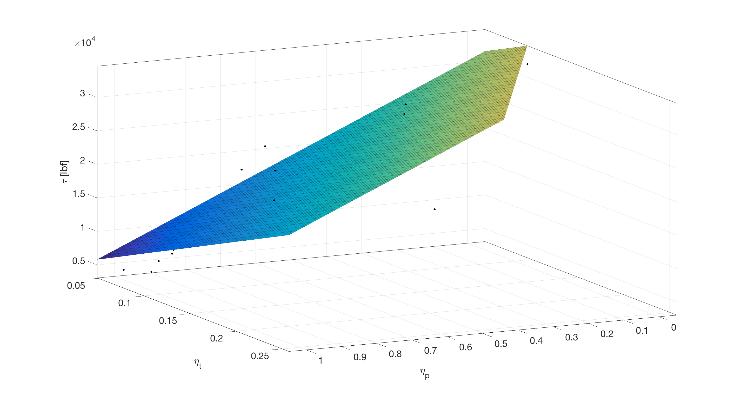


Figure 33:Thrust vs Efficiencies 3D Render

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|  |  |
| --- | --- |
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