

CubeSat Project Logbook

Team B

Claudio Vestini

Common part

Team members

Claudio Vestini

Alex Berresford

Fizza Naqvi

Hani Moussa

Code of Conduct

This Code of Conduct establishes guidelines for behaviour and collaboration among members of the [Project Name] group. We aim to create a respectful, inclusive, and productive environment for all participants.

Please continue from here.

Summary of the project and objectives

This project...

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2024-21-10 First meeting

PRESENT: CLAUDIO VESTINI, HANI MOUSSA, ALEX BERRESFORD, FIZZA NAQVI

APOLOGIES: NONE

LOCATION AND TIME: RSL STUDY ROOM 4 AT 14:00

AUTHOR OF MINUTES: CLAUDIO VESTINI

- Discussion of project organisation:
 - File system (GitHub repository, GitHub Projects roadmap (Gantt chart))
 - Google Drive folder
 - Report LaTeX file
 - References (.bib master file)
 - Meetings and WhatsApp group for communications
- Allocation of tasks (initial draft):
 - Claudio:
 - Aerothermal
 - Instrumentation
 - Hani:
 - Electronics
 - Interfaces
 - Fizza:
 - Trajectory
 - Internal heat generation
 - Alex:
 - Mechanical
 - Launch service provider
 - Launch environment
- Discussion of scientific goals:
 - CubeSat constraints dictated by launch service provider (size, weight, center of mass, electronics, stress response) - Alex
 - Ionospheric disruption due to re-entry impact - Fizza
 - Consideration of Magnus Effect during hypersonic re-entry – Alex
 - Budget analysis - everyone
 - Model Predictive Control for maintaining trajectory attitude (both in orbit and during re-entry). Use of cold gas thrusters as actuators - Claudio
 - Black box (GPS-tracked, ablative-protected) for retaining re-entry data – Alex
 - Materials testing for re-entry – Hani
 - Communications: information transfer during blackout – Claudio
 - Modelling the aerothermal environment in different re-entry stages - Claudio

References

Actions

- Discuss scientific goals with supervisors

Deadlines

2024-22-10 Second meeting

PRESENT: ALEX, CLAUDIO, HANI, FIZZA, TOBIAS (SUPERVISOR)

APOLOGIES:

LOCATION AND TIME: LR7 AT 2:00PM

AUTHOR OF MINUTES: ALEX BERRESFORD

- BRIEFING TOBIAS ON OUR PROGRESS, FILE SYSTEM, ORGANISATION ETC
 - MENDELEY FOR .BIB FILE FOR AUTOMATICALLY REFERENCING PAPERS
- BRIEFING TOBIAS ON PROJECT IDEAS
 - IONOSPHERE DISTURBANCES
 - FEEDBACK: INTERESTING, BUT A BIT OF A SECONDARY GOAL, NOT DIRECTLY RELATED TO RE-ENTRY
 - MATERIALS FOR RE-ENTRY
 - USE CUBESAT AS A TEST RIG FOR MATERIALS AND HOW THEY DEMISE IN EXTREME FLOW CONDITIONS
 - FEEDBACK: ON TOPIC, VERY CURRENT BIT OF RESEARCH FOR SPACE INDUSTRY
 - HOW WOULD YOU MITIGATE INEQUALITIES IN MATERIAL CONDITIONS
 - SAMPLE SPHERE'S INSIDE SACRIFICIAL SHELL?
 - ALTITUDE CONTROL USING SPIN
 - MAGNUS EFFECT
 - FEEDBACK: COULD BE USED TO CONTROL MATERIAL CONDITIONS TO ALLOW FOR TESTING
 - SERIOUS CONTROL PROBLEM

OVERALL FEEDBACK:

- FIND ROUGH BOUNDS TO PROBLEM THROUGH RESEARCH AND ROUGH CALCULATIONS
- Budget unlimited, but must be justified
- Black box vs Comms system
 - Both realistic, depends on specific design choices

Long Term goals

References

Actions

Deadlines

Research Tasks by 29/10/2024

- Hani – sensors for material degradation
- Claudio – Magnus effect, and realism of generating spin
- Fizza – Look into trajectory, expected burn altitude and ideal orbital altitude as well as ionosphere
- Alex - Investigate different cubesat geometries, costs, pros, cons et. Keep up with Launch provider research.

2024-28-10 Third meeting

PRESENT: CLAUDIO VESTINI, HANI MOUSSA, ALEX BERRESFORD, FIZZA NAQVI

APOLOGIES: NONE

LOCATION AND TIME: RSL STUDY ROOM 2 AT 13:00

AUTHOR OF MINUTES: FIZZA NAQVI

- Discussion on how to get Mendeley working for references
- Hani's research: discussion on the different types of sensors that already exist
 - Accoustic emission sensor
 - Recession sensors (used to measure how thermal protection systems are damaged as they enter the atmosphere); NASA and ESA has used this before so there's lots of information available
 - Look into what we're actually going to measure before deciding on what sensors we should use
 - Ensure that our experiment cannot be easily conducted on Earth
- Claudio's research: magnus effect and MPC
 - Magnus effect at hypersonic speeds works very differently
 - Most research is done on sphere's but calculations might be able to be manipulated to work with a cube
 - Looking at simulations- the ones that are currently available are limited as it won't test everything we need
 - Magnus effect can be tested when we have our CAD models
 - For control: our main options are cold gas thrusters
 - Reaction wheels- cheapest, easiest to manufacture, least risk involved but takes up lots of space, quite heavy
 - other forms of thrust such as hypergolic- mainly used in thrust systems in capsules or small satellites; easy however it's extremely toxic; slightly more expensive
 - MPC
 - Find a company that has architecture already made up for this or make it from scratch
 - We need 2 separate controllers
- Fizza's research:
 - Burn up altitude is typically 80-120km but depends on size, mass orientation and material composition
 - Design for Design study- use semi controlled re-entry
 - Trajectory model that simulated Cubesat re-entry trajectory; lots of assumptions are made on the atmosphere calculations and dynamic calculations
 - Ionospheric impact research- the range at which satellite demise occurs overlaps with the "E region" which reflects radiowaves and is essential for long distance communication

- Could monitor atmospheric composition changes because materials from the cubesat could remain in the ionosphere temporarily- use spectrometers to detect the wavelengths and see how the different material affects the ionosphere composition, therefore radio wave reflection and long distance communication
- Alex's research:
 - NASA has info on different possible cubesat sizes- we want to do a 1U size due to how easy the geometry is, but we could expand greater if needed
 - Endurosat- cost calculator; limited to a 1.5U platform

References

Actions

- Ask Tobias about what data would be good for our measurements

Deadlines

2024-29-10 Fourth meeting

PRESENT: ALEX, CLAUDIO, HANI, FIZZA, LUKE (SUPERVISOR)

APOLOGIES: NONE

LOCATION AND TIME: LR7 AT 2:00PM

AUTHOR OF MINUTES: HANI MOUSSA

- Discussion of mission (material testing for hypersonic re-entry)
 - Recession sensors/Acoustic emission sensors
 - Experimental use of sensors is viable if well-researched
 - Acoustic environment information could be researched
- Thrust for deorbit
 - Low orbits will be brought in by drag
 - Active re-entry is likely more practical
 - Consider price/how well-established each technology for thrust is
 - Ion thrusters are for longer missions
 - Cold gas thrusters may be more practical/cheaper
- Launch Service Provider
 - Can get in touch with providers/external companies/physics department
 - Be upfront and professional
 - Can get basic information on launch costs
 - Materials not easily comparable between companies
- Model Predictive Control
 - Model needed for cube tumbling into atmosphere
 - Relation to materials testing
 - Initial idea - even tumbling on all sides
 - Speed of trajectory/speed of tumbling need to be considered relatively
- Possible secondary mission objectives
 - Magnus effect in orbit
 - Ionosphere experimentation
 - Difficult to measure through the atmosphere
 - Good to look at environmental effects of satellite demise
- Transmitting data
 - Blackbox/Comms system options
 - Formalise choice process/create spreadsheet and compare qualities
 - Quantity of data
 - Rate of data
 - Likelihood of survivability
 - Price
 - Justification should be in logbook and report
 - Can carry out a similar process for sensors
- Originality of design
 - Use necessary qualities of product to pick items off the shelf
 - Microcontrollers/thrusters etc.
 - Need to be space-certified or need to be tested (legislation side of things)
- Deciding next steps

- Need to add numbers to decisions
- Batteries and reaction wheels
- Comms/Blackbox
- Mass limit and Budget need to be considered

References

Actions

- Alex - Re-entry breakup (Blackbox system), cold gas thruster comparison
- Claudio - Spin rate vs re-entry rate, motors needed for reaction wheels and their weight
- Fizza – Ionosphere measurement specifics, background trajectory information
- Hani - Compare possible options for sensors in more depth
- Long term considerations – get in contact with relevant companies for information

Deadlines

2024-04-11 Fifth meeting

PRESENT: CLAUDIO, ALEX, FIZZA, HANI

APOLOGIES: NONE

LOCATION AND TIME: RSL AT 5PM

AUTHOR OF MINUTES: CLAUDIO

Content goes here

- Alex – re-entry system:
 - Blackbox Idea not going to work due to weight restrictions, 4.0 kg + housing -> 8.6kg
 - Thrusters: factsheets -> possible choices (not clear, contact companies):
 - 300g mass, 100uN to 10mN thrust – hydrazene
 - HPGC thruster – low toxicity, low freeze point, 40g mass (no nozzle),
 - Batteries:
 - Optimus 30: large dimensions, 268g 30wHR
 - B14 modular: 375g, 45Whr, no NASA certification
- Fizza:
 - Ionosphere:
 - studies by ESA, cannot be used as classified
 - Remote sensing – companies:
 - Ground-based: higher resolution, no data storage problem
 - Balloons: difficult, coordination complexity, path complexity, time complexity
 - Justification of secondary objective due to regulations
- Hani:
 - Sensors:
 - Spreadsheet of several sensors for comparison:
 - Recession sensors not readily available – emerging technology, could build ourselves or contact ESA for purchase
 - GENERAL POINT: if price is not available, estimate in report
 - RSComponents website (not made for space, but cheap and used in the past in space applications), could lower price significantly
 - Papers: types of sensors used in projects – thermocouples (light, cheap, use several), mosaic core (infrared camera, not made for space so not certain we can certify it, 21mm largest dimension – viable (used in cubesats in the past))
 - Can we certify things that have not been certified for space? (ASK TOMORROW). How do we design tests.
 - Could be the case that we do not need to be as rigorous with certification as it is only necessary if you stay in atmosphere for a long time - > our satellite demises so could get away w/o certification if launch company is okay with it -> Ask someone at the company

- Claudio:
 - Book for general understanding of hypersonic regimes, for both trajectory and aerothermal environment – relations can be found nicely displayed in graphs
 - Mass of typical re-entry attitude control system below 200g – very slow rotation rates and very weak forces. Ditched idea of controlling during re-entry but could easily spin up using loads of time to do so before hitting atmosphere
 - Paper on reaction wheels design and modelling -need 3 of them
 - Found a paper on the design of a reaction wheel-controlled cubesat – very useful as it contains lots of pictures and cad files of the architecture – should use as reference when designing our own satellite (BEESAT)
 - Paper on empirical results of hypersonic testing of cubesat topologies.
 - Roshko number, Strouhal number, paul bruce -> tumbling objects for re-entry
 - Youtube videos for hypersonics CFD simulations

References

BEESAT: A Pico Satellite for the On Orbit Verification of Micro Wheels

Actions

- Alex: document choice of no black box
- Fizza: document choice of ionosphere effects as secondary objective, document choice of ground sensing (why are alternatives not viable?)
- Hani: decide on recession sensors
- Claudio: look at thermal transfer rates for different spin rates

Deadlines

2024-05-11 Sixth meeting

PRESENT: ALEX, CLAUDIO, FIZZA, HANI

APOLOGIES: NAME4

LOCATION AND TIME: 14:00 AT IEB LR7

AUTHOR OF MINUTES: ALEX BERRESFORD

Catching up Tobias on design choices

- Rule out Black box

- Settled for cold gas for altitude control

- Spectroscopy

 - Use Fibre coupled spectrometer (Thor labs), multiple fibres possible per spectrometer, one on each face is possible.

 - Ground observation difficult due to range.

- Space certification is on launch provider and not strictly legislative. Minimise risk where possible.

- Devices that will function in a space environment difficult to find:

 - Electronics want to be certified to ensure they won't be damaged by radiation.

 - Simpler components e.g. thermocouple/mechanical frame are more case by case.

- Problem obtaining technical components (e.g. recession sensors)

 - Make a mock up CAD and reference a paper describing use.

Based on component sizing, 1U design unrealistic.

Possibility of de-orbit using ISS "trash" system – Nanoracks deployment goes via ISS anyway. – solves deorbit issue.

Spin up in vacuum during de-orbit but before colliding with atmosphere to avoid competing with aerodynamic forces.

 - Dependent on launch provider altitude.

Roshko number – ND group for describing oscillating flow mechanisms.

For electronics, heating needs to be critically considered. Build up models from 0D to having a heating solution.

 - Shielding should be considered for digital information stream to prevent bit flips, unnecessary for analogue streams.

References

Actions

Fizza – Design an orbit to allow for burn at apogee, followed by a spin up in vacuum before reaching atmosphere.

Hani - Background reading on heating for CubeSat electronic, followed by another look at thermocouple and recession sensor implementation.

Claudio - Roshko number, Strouhal number and CFD hypersonics.

Alex – begin CAD modelling to get idea of internal design.

Deadlines

RE-ENTRY BREAKUP RECORDER NON – VIABILITY

- NO COMMERCIALY AVAILABLE OPTIONS SUITABLE FOR CUBESAT SIZES
- DESIGNING ONE WOULD BE TOO TECHNICAL AND A PROJECT OF ITS OWN
- EXAMPLES OF DESIGNS FROM SPACE COMPANIES IN “BLACK BOX” FOLDER
- UNREALISTIC TO BUILD ONE SMALL ENOUGH FOR A CUBESAT THAT WILL SURVIVE RE-ENTRY
E.G. ESA DESIGN >8KG

SPECTROMETER OPTIONS INFO LINK

<https://www.thorlabs.com/thorproduct.cfm?partnumber=CCS100/M#ad-image-0>

2024-12-11 Seventh meeting

PRESENT: ALEX, CLAUDIO, FIZZA, HANI

APOLOGIES: NONE

LOCATION AND TIME: 13:30 IN HOLDER BUILDING

AUTHOR OF MINUTES: FIZZA NAQVI

FIZZA

- HOW FAR OUT WE NEED TO BE TO GENERATE ENOUGH SPIN TO GET INTO THE ATMOSPHERE
- SPAWNING THE CUBESAT TOO FAR OUT BURNS A LOT MORE ENERGY FROM GETTING THE 'SPAWN' PLACE TO THE ATMOSPHERE

CLAUDIO

- LOOKING AT THE KNUDSEN NUMBER AND MEAN FREE PATH; HOW THE INTERACTIONS OF PARTICLES CAN AFFECT THE TRAJECTORY
- CFD EXAMPLES THAT COULD BE USED WHEN WE HAVE CAD FILES
- STROUHAL NUMBER

HANI

- READING ON COOLING ELECTRONICS; DEALING WITH HEAT GENERATION FROM ELECTRONICS; SOME CUBESAT'S HAVE HEAT PIPES LINKED FROM COMPONENTS THEMSELVES TO THE OTHER COMPONENTS TO DEAL WITH THE HEAT
- PHASE-CHANGE MATERIAL – STORES LOTS OF ENERGY; COMMONLY USED FOR CUBESATS
- LOOKED INTO RECESSION SENSORS; WHAT MATERIALS WORK BEST (NICKEL)

ALEX

- START MAKING CAD FILES
- USED SOME EXISTING COMPONENTS AND MADE SOME FILES
- FOUND SOME REACTION WHEELS OF VARIOUS SIZES

DISCUSSION WITH LUKE:

- TREAT THE TRAJECTORY SIMULATIONS AS SEPARATE TO THE SPIN CALCULATIONS
- LOOK AT STEADY STATE MODELS, PERFORM CALCULATIONS
- IF FLOW SPEED AND SPIN SPEED TIME SCALES ARE EQUAL, THE SYSTEM ISN'T INTO STEADY STATE
- VALIDITY OF THE STEADY STATE CALCULATIONS/ANALYSIS
- TO CONSIDER THE THERMAL ENVIRONMENT OF THE ELECTRONICS, CREATE A HEAT TRANSFER FLOW ANALYSIS TO CONSIDER HOW HEAT TRANSFER AFFECTS EACH COMPONENT
- OBTAIN A SET OF EQUATIONS TO SOLVE WHAT THE STEADY STATE TEMPERATURE WOULD BE

DISCUSSION WITH TOBI:

- YOU WOULD NEED TIME-ACCURATE SIMULATIONS TO RESOLVE SOME OF THE TERMS, BUT THIS IS BEYOND OUR SCOPE
- USE A MATRIX METHOD TO DO THE HEAT TRANSFER ANALYSIS

- -GRID CONVERGENCE STUDY- TYPICALLY DONE WITH FEA AND CFD SIMULATIONS
- IN THE REPORT, INCLUDE FLOW CHARTS TO REPRESENT COMPLEX CODE INSTEAD OF DIRECTLY INCORPORATING THE CODE INTO THE REPORT

References

Actions

Hani- look at what temperatures the electronics can deal with; what does the heating scenario look like when simply being in orbit; look further into certain components such as battery choices and microcontrollers

Alex- email manufacturers for necessary CAD file components; work on CAD design

Fizza- Modelling and simulation of aerospace vehicles by Peter Zipfel; do some calculations on the required spin, distance, time, impulse of thrusters etc.

Claudio- look at the requirements for systems to be in steady state, quasi steady state, etc; continue CFD analysis

Deadlines

2024-18-11 Eighth meeting

PRESENT: ALEX, CLAUDIO, FIZZA, HANI

APOLOGIES: NONE

LOCATION AND TIME: 14:30 AT RSL

AUTHOR OF MINUTES: HANI MOUSSA

- TIMELINE DISCUSSION
 - LOGBOOK REVIEW NEXT WEEK – CLEAN UP
 - SPEAKER TOMORROW
- HANI'S MICROCONTROLLER/BATTERY CHOICE
 - LIST OF COMMON PROCESSORS ON CUBE SATS
 - MANY POSSIBLE OBCS
 - SPECIFIC DECISIONS DEPENDANT ON MISSION REQUIREMENTS
 - BATTERY MATERIAL TYPES
- ALEX'S COMMUNICATION WITH SUPPLIERS
 - REJECTED INFORMATION REQUEST FOR PROPULSION SYSTEM
 - MODULAR, CUSTOMISABLE COMPONENT DEPENDANT ON CUSTOMER REQUIREMENTS
- POSSIBLE COLLISION
 - AVOIDABLE WITH REACTION WHEELS/PLANNING/THRUST
- FIZZA'S TRAJECTORY CALCULATION
 - STARTING AT 400KM (ISS LEVEL), SPINNING UNTIL DEORBIT BURN (250KM)
 - ALTITUDE CONTROL COULD BE DONE WITH THRUSTERS – WOULD NOT REQUIRE HIGH MASS (~1 GRAM)
 - HARDER TO DESIGN THAN REACTION WHEELS
 - RESEARCH AVAILABLE FOR MATHEMATICS OF REACTION WHEEL USE
 - STABILITY REQUIRES LOW FREQUENCY (1HZ ORDER OF MAGNITUDE)
 - THRUSTER REQUIRED NOT TO AFFECT SPIN
 - DEORBIT THRUST COULD OCCUR BEFORE SPIN
 - IF SPIN THRUST COMES FIRST, TIMING MAKES A HARDER PROBLEM
- MAGNUS EFFECT
 - SPIN IS SLOW FOR MAGNUS EFFECT
- RE-ENTRY TIMELINE AND SIZING CONSIDERATIONS
 - RE-ENTRY BURN, ATTITUDE ACTIVATION, BURN UP
 - CONTROL FOR 3U CUBE SAT
 - STABLE RE-ENTRY AIDED BY POSITIONING OF CENTRE OF MASS
 - ENTRY SURFACE CAN BE ONE OF THE SMALLER FACES IF SPINNING AROUND LONGER AXIS
 - ALTERNATIVE RE-ENTRY SURFACE AND SLIGHTLY MISALIGNED CENTRE OF MASS CAUSES UNINTENDED SPIN
 - THERMAL EQUILIBRIUM NOT REACHED FOR MATERIALS TESTING
 - LARGER SATELLITE CONSIDERABLE?
 - 8U WOULD BENEFIT THE MATERIAL TESTING EXPERIMENT
 - LARGER SATELLITE MAY REQUIRE HIGHER BUDGET
 - SPLIT 3U INTO 1U DETACHMENT FOR MATERIAL TESTING EXPERIMENT
 - ADVANTAGES
 - SIMPLIFIES DESIGN FOR 1U SECTION

[DATE][SIGNATURE]

- DISADVANTAGES
 - DETACHMENT IS DIFFICULT (WIRING/BATTERIES/SIDE OF 1U)
 - POSITIONING OF COMPONENTS IS DIFFICULT
 - TRAJECTORY WILL BE AFFECTED
 - 1U CUBESAT
 - THEORETICALLY POSSIBLE, BUT FITTING EVERYTHING MAY BE POSSIBLE
 - LAUNCH MAY BE EXPENSIVE
 - LAYERED DESIGN AS IN BEESAT
- CLAUDIO'S RESEARCH ON AERODYNAMICS SITUATION
 - THERMAL LOAD/VELOCITY STREAM ON EXAMPLE SATELLITE
 - CFD RUNS
 - STROUHAL NUMBER HAS A LOW ORDER OF MAGNITUDE WITH LOW FREQUENCY
 - TIME TO GO BETWEEN STEADY STATES IS VERY LOW
 - ALLOWS ASSUMPTION OF CONSTANT STEADY STATE

References

Actions

Deadlines

2024-19-11 Ninth meeting

PRESENT: ALEX, CLAUDIO, FIZZA, HANI

APOLOGIES: NONE

LOCATION AND TIME: 14:00 AT IEB LR7

AUTHOR OF MINUTES: HANI MOUSSA

Belstead Re-entry Talk

- Destructive re-entry
 - Some debris can survive
- Uncertainties
 - Aerothermodynamics
 - Thin parts get hot first (titanium bipod test)
 - Calculations are not necessarily strong predictors, testing required
 - Fragmentation
 - Electronics box
 - Housing fails
 - aluminium warps under oxide layer influence
 - steel pins survive longer
 - electronics card survives past metals
 - Material Response
 - Liquid droplets, oxide layers on stainless steel
- Knowns
 - Demise qualities
 - Continuum heating dependant on length scale
- Unknowns
 - Rarefied heating
 - Structure failure mode in re-entry
 - Materials responses to failure
 - Metals
 - Ceramics
 - Composites
- QnA
 - Predictions
 - Speed/air density/size define drag/heating
 - Use literature
 - High up for CubeSats
 - Box of doom
 - Tumbling
 - Tumble-averaging heat flux, thermal approximation
 - Numerical extrapolation

Experiment assessment

- Fragmentation causes casualty risk
- Experiments to this end

[DATE][SIGNATURE]

- EntrySat
 - Qarman
- Flight recorder
 - Transmits after blackout
 - Parachutes/buoyant
 - Difficult to apply to CubeSat
- Dedicated vehicles
 - Qarman survives blackout
 - Heatshield
 - Aerodynamically stable
 - VAST + VASP
 - Large vehicles
 - Thermally insulated electronics
 - Not applicable to 3U
- Measurements
 - Images and video are very helpful
 - Not necessarily high resolution
 - High number of low res >> low number of high res
 - Thermocouple/pressure traces aren't helpful by themselves
 - Images are data hungry, however
 - Thermocouple data high priority
 - Doesn't require high data rates
- Repeatability
 - Demise behaviour may vary from CubeSat to CubeSat
 - Repeatable CubeSat is very valuable – allows consistent scientific results
- QnA
 - Blackbox idea
 - Great in theory
 - Issue is lack of volume in a CubeSat
 - Difficulty of transmitting data
 - Transmit through radar-transparent material
 - Spherical sat (e.g. iball) has wide ability to transmit
 - Aerodynamically stable sat allows simple transmit direction
 - IRIDIUM satellite network

Discussion with Luke

- Don't expect us to solve every problem
 - $\geq 50\%$ expectation of working
- Transmitting information
 - No spin allows transmitting out the back of the satellite
 - Tumbling could use multidirectional antenna
- Size
 - Smaller = simpler
 - Size decision (1U) allows boundaries for power/size/cost
- Materials not possible on every side due to size constraint
 - Could have material for testing on not every side/on 80% of sides

- Timeline
 - Current idea as described in yesterday's meeting
 - Transmission requires radio-transparent materials
- Transmission
 - Tumbling limits window of transmission for single-direction antenna
 - Side panel with unidirectional antenna not part of material experiment
 - Choice comes down to data-rate required/instrumentation
- Mission objective
 - Secondary objective is beneficial to materials testing customers – track environmental impact
 - Spectrometer is large for 1U, would work for 8U
- Sensor on outside
 - Glue – easy to take off
 - Solder – wire will be broken down
 - Bore-hole – measure under the surface, but doesn't measure true surface temperature
- Logbook review next week
 - Go over logbooks
 - Tidy up logbooks till then

Not examinable till end of project

References

Actions

Deadlines

2024-11-26 Tenth meeting

PRESENT: ALEX, CLAUDIO, FIZZA, HANI

APOLOGIES: NONE

LOCATION AND TIME: LR7 AT 15:00

AUTHOR OF MINUTES: ALEX BERRESFORD

Presentation sum up from Tobias

- Total 20 minute presentation
 - ~ 5 minutes each
 - cohesive, not 4 individual presentations
- Give enough detail so the audience knows what's going on and can make a judgement
- In general pitch to audiences understanding
- For undecided options, present both and give a conclusion to how that decision will be made.
- Referencing
 - Ideally on the slide, abbreviation ok
 - Sum up references on final slide alongside abbreviated on slide referencing
- Not formally assessed, purely for feedback
- OK to present present work done, not based on pure calculations e.g. for mechanical/electrical
- Present on work done

How detailed should trajectory calculation be after re-entry?

Velocity and force balance every ~ 0.5km

Google slides vs Beamer vs Powerpoint Online

- Beamer decided as it will develop useful Latex skills

1U vs 8U

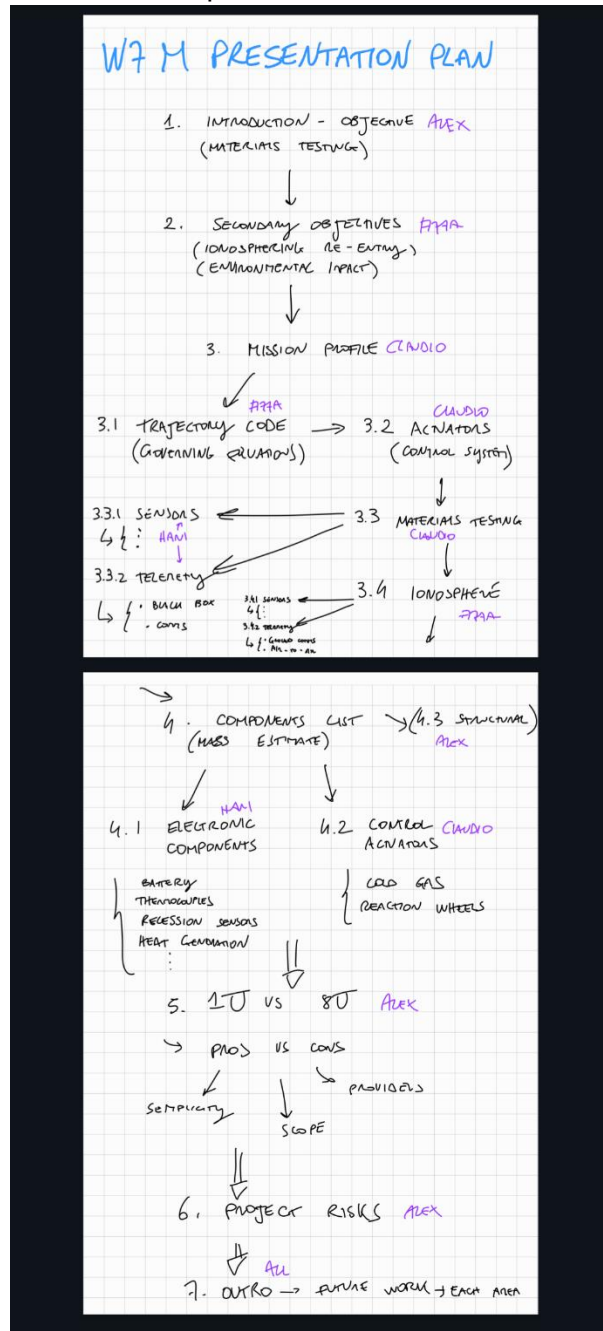
- 1U is simpler and far cheaper for materials testing rig
- 1U has very limited volume limitations
 - Diagonal OBC and battery?
- 8U still cube for tumbling
- 8U allows space for secondary objectives, e.g. Ionosphere
- No packing problem

Sum up chosen components in spreadsheet for mass estimate of prototype

Presentation for next Tuesday 2nd Dec – meeting Friday 29th Nov

- Begin with primary objectives
- Give cohesive, continuous presentation
- 1U vs 8U “debate” heavily featured

More detailed plan below



References

Actions

Everyone to prepare content for their assigned slides – see plan by the next meeting – 29/11/24

Deadlines

2024-11-26 Eleventh meeting

PRESENT: ALEX, CLAUDIO, FIZZA, HANI

APOLOGIES: NONE

LOCATION AND TIME: TEAMS (ONLINE), 13/12/2024 AT 10AM

AUTHOR OF MINUTES: FIZZA NAQVI

Decision made to go with 8U

- allows us more freedom to do “more engineering”
- easier to design/fit components in
- materials on the outside can be thicker

Decision needs to be made on which materials to use

- look at ablatives typically used by other companies
- think about how long it would take certain materials to break down

Communication

- Look at where we can and can’t communicate properly
- Potentially reconsider blackbox idea
- With the ablatives on the outside, it may be difficult to communicate through the materials
- Potentially have an antenna on most of the faces
- Biggest project risk- not getting data

Power

- Number of batteries/type of battery depends on how much power we will consume

Actions

Alex

- Look at the vibrational model

Fizza

- GMAT simulation

Claudio

- Look at the materials

Hani

- Look at the comms and what can be done