

# **Power Distribution System for a CubeSat**

*Project report  
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in partial fulfillment of the requirements  
for the degree of*

**Bachelor of Technology in Electrical and Electronics Engineering**



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## DECLARATION

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We, the undersigned, hereby declare that the mini project report titled *Power Distribution System for a CubeSat*, submitted for partial fulfillment of the requirements for the award of degree of Bachelor of Technology of the APJ Abdul Kalam Technological University, Kerala is a bonafide work done by us under the supervision of Dr. Dinesh Gopinath, Department of Electrical and Electronics Engineering, Government Engineering College, Barton Hill. This submission represents our ideas in our own words and ideas and words of others have been included, we have adequately and accurately cited and referenced the original sources. We also declare that we have adhered to ethics of academic honesty and integrity and have not misrepresented or fabricated any data or idea or fact or sources in submission. We understand that any violation of the above will be a case for disciplinary action by the institute and/or the University can also evoke penal action from the sources which have thus not been properly cited or from whom proper permission has not been obtained. This report has not been previously formed the basis for the award of any degree, diploma or similar title of any other university

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**CERTIFICATE**

This is to certify that the report titled **Power Distribution System for a CubeSat** submitted by **Ansaf Niyaz, Govind Murali, Jijesh J. Kumar, Naveen A.B.** of the **Department of Electrical and Electronics Engineering** to the APJ Abdul Kalam University in partial fulfillment of the requirements for the award of the Degree of *Bachelor of Technology in Electrical and Electronics Engineering* is a bonafide record of the project work carried out by them under my guidance and supervision. This report in any form has been submitted to any other university or institute for any purpose.

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## **ABSTRACT**

CubeSats are miniature versions of satellites that offer hands-on experience to engineering students in designing, developing, testing and operating a real space-craft system. A 1U CubeSat is a cube shaped satellite with dimensions of 10 cm x 10 cm x 10 cm and maximum mass of 1.33 kilograms. CubeSats are traditionally built from COTS-components (Commercial Off-the-Shelf) with low resources. Typically, CubeSat have limited mission time and short development and testing time.

One of the most critical aspects of the CubeSat is the Electrical Power System (EPS) since the electrical power is necessary for a CubeSat to operate. The EPS of the CubeSat consists mainly of solar cells, batteries, voltage converters and protection circuits. The EPS is responsible of providing stable power to the CubeSat subsystems.

The purpose of this project is to design and implement an EPS for a CubeSat. The EPS must be able to power all subsystem components including telemetry, on-board computer, attitude determination and control system , thermal system as well as the payload while also protecting the subsystems from the over-current and over-voltage issues associated with the device failure. The system will be designed to provide power for the satellite throughout the entire orbit, even during periods of eclipse when the satellite is not able to generate power. The EPS should also provide data about voltage and current measurements, battery status, etc. to OBC (On-Board Computer).

## **ABBREVIATIONS**

|      |   |
|------|---|
| DC   | Direct current                            |
| IC   | Integrated Circuit                        |
| EPS  | Electrical Power System                   |
| MCU  | Micro-controller Unit                     |
| OBC  | Onboard Computer                          |
| PCB  | Printed Circuit Board                     |
| PWM  | Pulse Width Modulation                    |
| RBF  | Remove Before Flight                      |
| TCS  | Thermal Control System                    |
| TTC  | Telemetry, Tracking and Command           |
| ADCS | Attitude Determination And Control System |
| MPPT | Maximum Power Point Tracking              |
| OCP  | Over Current Protection Circuits          |

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# Chapter 1

## Introduction

### 1.1 Background

Artificial satellites are foundational components of modern society. Satellites have played a huge role in communication, navigation, scientific research, military surveillance and the study of the solar system and beyond.

Artificial satellites are launched into space using rockets and are placed in orbit around the Earth or other celestial bodies using thrusters. Once in orbit, they remain in space and continuously circle around the Earth, following a specific path called an orbit.

There are two main types of orbits for artificial satellites: geostationary orbit and low Earth orbit. Geostationary orbit is a circular orbit located at an altitude of approximately 36,000 kilometres above the Earth's equator. Satellites in this orbit appear to be stationary relative to a fixed point on Earth, making them ideal for communication and navigation purposes. Low Earth orbit, on the other hand, is an orbit located at an altitude of up to approximately 2,000 kilometres above the Earth's surface. Satellites in this orbit circle the Earth more frequently and are used for scientific research, remote sensing, and military surveillance.

Artificial satellites are powered by solar panels that convert sunlight into electricity, which is used to operate the satellite's systems and instruments. They are also equipped with communication antennas that enable them to transmit and receive signals from Earth or other satellites.

The first artificial satellite, Sputnik, was launched in 1957. The launch of the first meteorological satellite, TIROS-1, in 1960 was followed by the U.S. The first Earth-observing satellite for land-based applications was Landsat-1, which was

launched in 1972. Since then many more Earth-observing satellites have been put into orbit. Overall, artificial satellites have revolutionized our ability to communicate, navigate, observe, and study our planet and the universe beyond. They have become an essential part of modern life, and their applications are only expected to increase in the future.

The whole process of designing, building, and launching a satellite-flown remote sensing system is a very lengthy and costly process. The main driving force behind the enormously costly development of space technology has been the military, where cost is not usually a problem. With the relaxation on restrictions for ownership and operation of Earth-observing satellites it is now possible for anyone to build and launch satellites.

For more than a decade, CubeSats, or small satellites, have paved the way to low-Earth orbit for commercial companies, educational institutions, and non-profit organizations. These small satellites offer opportunities to conduct scientific investigations and technology demonstrations in space in such a way that is cost-effective, timely and relatively easy to accomplish. Due to their small size, CubeSats can be launched on smaller rockets or piggybacked on larger missions, allowing for more frequent and flexible deployment opportunities. The small size of CubeSats allows for faster development and testing of new technologies, which can be later applied to larger satellites or space missions.

A CubeSat is a class of miniaturized satellite based around a form factor consisting of 10 cm cubes. CubeSats have a mass of no more than 2 kg per unit, and often use commercial off-the-shelf (COTS) components for their electronics and structure. CubeSats are put into orbit by deployers on the International Space Station, or launched as secondary payloads on a launch vehicle. As of August 2021, more than 1,600 CubeSats have been launched.

CubeSat missions benefit Earth in varying ways. From Earth imaging satellites that help meteorologists to predict storm strengths and direction, to satellites that focus on technology demonstrations to help define what materials and processes yield the most useful resources and function best in a microgravity environment,

the variety of science enabled by CubeSats results in diverse benefits and opportunities for discovery.

Electrical power systems (EPS) are an essential component of CubeSats as they have a limited power budget and often require a reliable source of power to operate their payloads and subsystems. There are several ways to design an EPS for a CubeSat, and the choice depends on the mission requirements and the available resources. The most common EPS design for CubeSats is based on solar panels, which convert sunlight into electrical energy.

The solar panels are typically connected to a power management and distribution unit (PMDU), which regulates the voltage and current of the incoming power and distributes it to the various subsystems and payloads. The PMDU also includes a battery that stores excess power generated by the solar panels and provides power during eclipses or when the solar panels are not receiving enough sunlight.

The EPS also includes other components such as a power switch, a fuse, and a battery charge controller. The power switch is used to turn on or off the power to the subsystems and payloads, while the fuse protects against over-current and short circuits. The battery charge controller manages the charging and discharging of the battery, ensuring that it is not overcharged or over-discharged.

The design of the EPS depends on the specific mission requirements and the available resources, and CubeSat developers must carefully balance the power budget to ensure that the CubeSat can operate reliably and efficiently throughout its mission.

## 1.2 Objective

The aim of this project is to determine requirements for a typical CubeSat Electrical Power System (EPS) and develop a working prototype of the EPS for a CubeSat.

This EPS will be used as the prototype for the BARTOSAT cubeSat developed by the students of GEC, Barton Hill as a part of the Satellite Research Center of the college. The prototype will then go through various iterations, improving the design to meet all the requirements of other subsystems which are currently under

development and also to make it fully capable to work in the space environment.

### **1.2.1 The Satellite Research Center, GECB**

The Satellite Research Center of Government Engineering College, Barton Hill, is a facility dedicated to conducting research in the field of satellite technology and to help students get an exposure to space technologies in general. As a part of this, a student project was initiated, aimed at developing a 1U CubeSat named 'Bartosat', which will be designed to carry a LoRa module as the payload.

Every phase of designing, building and testing of the CubeSat has been carried out by different student teams from various departments. The payload will test the possibilities of space-to-ground communication using LoRa technology, which can lead to low cost, low-power space communication.

## **1.3 Literature Review**

The EPS is a critical component of a CubeSat as it powers all the subsystems in the satellite. There are different EPS architectures based on DC bus regulation and interface of PV panels [1].

The main challenge for the development of CubeSats is to fit all the necessary equipment within the standard frame size while meeting the weight constraints. The EPS design is crucial for the successful CubeSats mission, which needs to consider several factors such as mission duration, orbit altitude, inclination angle, size of PV panels and its arrangement, load profile, volume and weight limits, radiation effects, overall efficiency, simplicity of control, component count, flexibility in battery configuration, reliability, and fault-tolerant capability.

One of the important steps in EPS design is the selection of most suitable EPS architecture. The different types of CubeSat EPS architectures are shown in Figure.

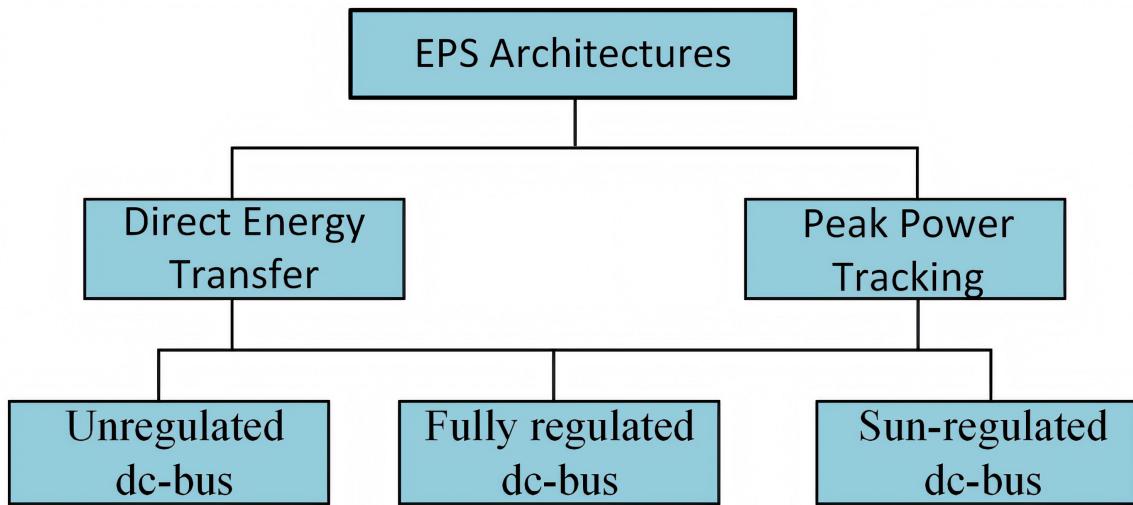


Figure 1.1: CubeSat EPS architectures

Based on the interfacing of PV panels, the EPS architectures are classified as direct energy transfer (DET) and peak power tracking (PPT) architectures.

In the DET architecture, the PV panels are directly connected to the battery and/or load equipment via diodes. It also uses a shunt regulator in parallel with the PV panels to absorb excess power when the battery reaches full-charge condition.

In the PPT architecture, the PV panels are interfaced with a power electronic converter to extract the maximum power from the panels under widely varying operating conditions such as solar irradiation, PV panel temperature, and inclination angle.

The DET architecture requires matching of PV panel characteristics and the dc-bus voltage to generate maximum possible power which is not straightforward in the space missions. Therefore, the majority of CubeSats utilize the PPT architecture to maximize the solar energy harvest, which has limited PV panel capacity and storage capacity due to strict volume and weight constraints.

The PPT architectures are further classified based on dc-bus voltage regulation: unregulated dc-bus, regulated dc-bus; and sun-regulated dc-bus.

The unregulated dc-bus EPS architecture has the battery terminals directly connected to dc-bus. When the battery voltage reaches its upper limit, the MPPT converter is operated in voltage regulation mode to avoid further charging of the battery.

The regulated dc-bus EPS architecture interfaces the battery with a dc–dc converter to maintain the dc-bus voltage at the predefined value. This architecture enables the operation of CubeSat at higher dc-bus voltage so that conduction and ohmic losses are reduced.

In the sun regulated dc-bus EPS architecture, the dc-bus voltage is maintained at reference value only during the sunlit period. During the eclipse period, the battery gets connected to the dc-bus via a diode to supply the loads.

This project focuses on the design of an EPS with a regulated DC bus and MPPT tracking.

Based on the difference between centralized and distributed architecture was discussed in [2] and we have selected centralized architecture.

As discussed in [3], solar panels operate at their most efficient points with a power point tracking algorithm, allowing the extraction of maximum power from the solar panels. Hence, peak power transfer is preferred to direct power transfer.

Different battery technologies used in CubeSats were discussed in [4] and Li-ion cells were selected as the energy storage device due to their high energy density and higher number of charge discharge cycles compared to LiPo and NiMH batteries.

From [5], the optimum ambient temperature for charging a Lithium ion battery is +5°C to +45°C and thus, charging is limited to this range of temperature.

# **Chapter 2**

## **The Electrical Power System**

The Electrical Power System (EPS) is an electronic circuit board that is designed to supply, manage and store energy in an efficient way. The EPS must be able to harvest energy from the solar panels and store it in the battery, as well as delivering power to the satellite, using switch controlled converters to supply a regulated voltage. Redundant circuitry must be present to ensure continuous and reliable operation of the satellite in case of the failure of EPS components.

The output of the solar panels is first run through the power path control. While in sunlight operation, the power path will select the voltage from the panels based on its higher voltage. The output of the Power Path control is sent to DC-DC converters to provide 5V and 3.3V regulated DC supply for the CubeSat modules. During the eclipse, the power path will select the battery to power the circuit components. The software is implemented in order to manage the overall energy of the satellite, regulate the converters to extract maximum power from solar panels, perform power diagnostics, engage redundant circuitry and to communicate with the On Board Computer. The software also employs four operating modes: Initialization mode, Safe mode, Normal mode and Low Power Mode.

### **2.1 Requirements of EPS**

#### **2.1.1 Introduction**

The first step before beginning the design of the EPS is to know exactly what our goals and our constraints are. This is the subject of this section.

There are constraints due to space environment. The EPS must withstand vacuum and wide ranges of temperatures and radiations. There are constraints linked to the launch, like accelerations, vibrations, and rules for the CubeSats in the launch vehicle. Finally, there are constraints due to the construction of CubeSat itself. The dimensions of the PCB and available volumes should be specified.

The desired functionalities also guide the design. The EPS must produce enough energy to supply the CubeSat. Enough energy has to be stored to supply the satellite during the period of eclipse. The EPS must provide several power outputs with stabilized voltages. The constraints will also guide the validation tests applied to the prototypes. Since our objective is to make demo boards, all these constraints may not be fully follower while designing.

### **2.1.2 Constraints**

#### **2.1.2.1 Vacuum**

The satellite is expected to be released in an elliptical orbit with an altitude between 350 and 1,000km. At such altitudes, the atmosphere pressure can be neglected and considered as vacuum. Therefore, all components used in the satellite must withstand vacuum. The most sensitive component of the EPS is the batteries. The two threats of vacuum are:

- Deformations due to mechanical constraint of vacuum
- Out gassing.

#### **2.1.2.2 Radiations**

Charged particles of solar wind, electrons, and protons, are captured by the earth magnetic field. They form the radiations belts, also known as the Van-Allen belts. There are two belts:

- The inner belt, between 1,000 and 15,000 km, containing high concentrations of energetic protons with energies exceeding 100 MeV and electrons in the range of hundreds of keV.

- The outer belt, extending till 50,000km, and consisting mainly of high energy electrons (from 0.1 to 10 MeV).

Spacecrafts need to be protected against radiations, especially if they go through the radiation belts. Trapped particles in the radiation belts and cosmic rays can cause “Single Event Phenomena” (SEP) within semiconductor devices. There are three different types of SEP:

- The Single-Event Upset (SEU)
  - This is when a high energy particle hits a logic device and changes digitally stored data or causes a gate to open or close at the wrong time.
- The Single-Event Latch up (SEL)
  - The SEL is when a high energy particle directly damages the device. It can, however, be corrected if the SEL is detected and the power to the device quickly turned off, then turned back on.
- The Single-Event Burnout (SEB)
  - This is the case where the device is destroyed.

The radiation dose is estimated to be more than 105[rad]. A protection against SEL can be provided to the subsystems with current-limiter circuits. There is no particular protection against SEU and SEB except reducing the effect of the radiations inside the satellite with a layer of shielding aluminium (less than 2.104[rad] with 2mm of aluminium). There are components designed and/or tested to be more resistant to radiations. Such components should be used in the more critical systems of the EPS.

### **2.1.2.3 Temperature**

The temperatures in space, when the satellite is turned on, will essentially depend of the thermal design. The temperature ranges are not the same everywhere in the satellite. Thermal simulations give an idea of the temperature at different points of the CubeSat. The external temperature will vary the most (from  $-33^{\circ}\text{C}$  to  $+40^{\circ}\text{C}$ ), the temperature of the EPS will stay between  $-22^{\circ}\text{C}$  and  $+37^{\circ}\text{C}$ , and the temperature of batteries between  $-22^{\circ}\text{C}$  and  $+37^{\circ}\text{C}$ . The EPS must thus be able to work within these ranges.

- Solar cells must be selected so that they are able to work in the predicted range of  $-33^{\circ}\text{C}$  to  $+40^{\circ}\text{C}$ .
- The electronics on the EPS must be designed to be able to operate from  $-22^{\circ}\text{C}$  to  $+37^{\circ}\text{C}$  (PCB Temp). Considering wider ranges, components with an operating temperature range of at least  $40^{\circ}\text{C}$  to  $+85^{\circ}\text{C}$  can be selected as "absolute maximum rating".
- Lithium-Polymer batteries can withstand 0 to  $+45^{\circ}\text{C}$  during charge and  $-20^{\circ}\text{C}$  to  $+60^{\circ}\text{C}$  during discharge (but with a significant loss of capacity under  $0^{\circ}\text{C}$ ). A solution must be found to maintain the batteries in these ranges of temperature.

### **2.1.2.4 Vibrations and accelerations during launch**

The CubeSat will be subjected to vibrations during the launch. The effects of acceleration during launch are not to be underestimated. The CubeSat has to be able to withstand an acceleration of  $15\text{g}$ , even if it will probably be lower in reality. Special attention must be paid to the fixation of heavy components. Also, components could be damaged by the bending of the PCB under vibrations and accelerations. Automotive components will be chosen whenever possible.

### **2.1.2.5 Temperatures during launch**

The launch vehicle will pass through several atmospheric layers with specific temperatures. Inside the launch vechile, the CubeSat will endure temperatures of  $-40^{\circ}\text{C}$  to  $80^{\circ}\text{C}$  during launch. Components of the EPS have to be able to withstand

such temperatures during storage (the CubeSat is inactive during launch). Components with a working temperature range of  $-40^{\circ}\text{C}$  to  $85^{\circ}\text{C}$  should not have any problems. The batteries are still the most sensible component. Their storage temperature should stay between  $-20^{\circ}\text{C}$  and  $60^{\circ}\text{C}$ . A passive solution must be found to protect them during launch. Thermal insulation and thermal inertia will certainly help.

## 2.2 Components of EPS

The EPS of a CubeSat can be designed with many different architectures, but some components are common to all designs, such as:

- Solar panels to harvest the energy from the Sun
- Battery charger to manage the charging profile of the battery
- Voltage regulators to feed the regulated power bus of the satellite
- Remove Before Flight (RBF) switches and deployment switches, to cut the power while the satellite is not deployed

Other components of the EPS are:

- Battery and associated charging circuit
- Solar panels on 6 faces of the satellite
- MPPT converters which help optimise power collection from the sun
- Buck and boost converters which help provide required voltage busses for components of different voltages
- An MCU which controls the tasks that the EPS performs and monitors the status of the components
- Over Current Protection Circuit which helps protects important components from high current flow
- Current and Voltage sensors to keep track of their consumption.

- Temperature sensors to measure battery temperature, based on which battery heater is used
- Battery heater circuit

## 2.3 Tasks of EPS

Tasks of the EPS are:

- Collect housekeeping for various components associated with it, like the various current & voltage sensors and the battery's state of charge.
- Handle housekeeping requests and other commands from the OBC (ON/OFF requests of any subsystem by OBC).
- Implement MPPT to optimize power generation.
- Control the Simple Beacon (which contains only the call sign of the satellite) before the TTC gets switched on.
- Implement a watchdog timer to keep a check on the operation of the OBC.
- Take action on the basis of OCPC triggers.
- Deployment of antenna at the time of satellite initialisation.
- Turn on the battery heater when temperature goes below critical level

# **Chapter 3**

## **Methodology**

### **3.1 Identifying Power Requirements**

Before designing the EPS, the power requirements of the various subsystems of the CubeSat has to be identified. A power budget has to be prepared accounting all the energy, voltage and current requirements of the subsystems. The orbital parameters at which the CubeSat might be operating should also be considered. The orbital altitude, period and eclipse time and the daylight time has to be identified and documented. After this, the peak power budget has to be calculated and total energy and power demands are to be found out.

### **3.2 Literature Review**

In order to select the suitable architecture and topologies, literature study has to be conducted. Various articles regarding the implementation of CubeSats and EPS were studied and the findings were recorded.

### **3.3 Architecture Design and Topology selection**

The design of EPS starts with the selection of appropriate EPS architecture based on the comparison of overall efficiency, battery size, and reliability. The EPS design is critical for CubeSat mission success, therefore selection of proper EPS architecture is one of the important steps. Different standard EPS architectures are classified on the basis of various topologies like dc-bus voltage regulation, interface of

PV panels, location of power converters, and number of conversion stages. The necessary topology has to be selected based on the demands and constraints.

### **3.4 Forming Specifications**

After deciding upon a suitable architecture, the specifications of various components of the EPS has to be finalised. This includes deciding the number of required power converters and their input and output parameters, deciding the number, size and type of battery for energy storage and the characteristics of the solar panels and specifications of the MPPT device.

### **3.5 Design and simulation**

Suitable ICs able to perform the various functions of different components in an EPS have to be identified. The ICs must be suitable for operation in outer space. After selecting the ICs, the design of them are to be completed and necessary schematics and PCB design has to be completed. Also, the circuits obtained have to be verified with the help of simulation results.

### **3.6 Procurement of components**

The components which were finalised has to be procured. Surface Mount components are preferred due to the space constraints, also the selected components must be applicable in outer space applications.

### **3.7 Fabrication and Testing**

The components have to be soldered into the PCB and the results are to be observed. Initially, each component maybe developed individually and tested before optimizing the entire circuit into a single, centralized form.

# **Chapter 4**

## **Power Budget**

The power budget of a CubeSat is a critical aspect of its design and operation, as it determines the capabilities and limitations of the satellite. A well-designed power budget can ensure that the CubeSat operates efficiently and achieves its mission objectives, while a poorly designed power budget can lead to power shortages and mission failure.

It is important to determine the power budget at the beginning of the EPS design to determine the characteristics of the system. When the available space for the solar cells and the orbital parameters are known, power production can be estimated. The power requirements of CubeSat as a whole depend upon the power requirements of the individual components and how the components are used together for operations. Together with the efficiency information of the EPS components, this data is used to determine critical elements of the EPS design, like required solar array and battery size.

A CubeSat will have standard set of satellite subsystems: Structural subsystem, Telemetry, Electrical Power Subsystem (EPS), Thermal Control Subsystem (TCS), Attitude Determination and Control Subsystem (ADCS), On Board Computer (OBC) and Payload. For the calculations, a LoRa module was selected as the payload. The orbital parameters are given below:

| Parameter        | Value      |
|------------------|------------|
| Orbital altitude | 590 km     |
| Orbital radius   | 6968.14 km |
| Flight velocity  | 7.563 km/s |
| Orbital period   | 96.483 min |
| Eclipse time     | 31.164 min |
| Daylight time    | 65.319 min |

Table 4.1: Orbital Parameters

The CubeSat has an orbital altitude of 590 km with an orbital radius of 6968.14 km to maintain a flight velocity of 7.563 km/s. The orbital period is 96 min 29 sec with an eclipse time of 31 min 10 sec and daylight time of 65 min 19 sec. Based on the power budget, the energy required by the CubeSat is 1.997 Wh per orbit. Hence, the solar panels must be designed to produce at least 1.997 Wh per orbit.

Power requirements of various components of each subsystems are given below. Since the power production may vary due to parameters like efficiency of panels, margin and contingency are added to the total power requirements.

| Sub-system     |              | Voltage (V) | Max. Current (mA) | Power (mW) | Contingency 5%       | Margin 20%     | Duty Cycle (%)     | Energy (Wh)  |
|----------------|--------------|-------------|-------------------|------------|----------------------|----------------|--------------------|--------------|
| <b>ADCS</b>    | ADCS         | 3.3         | 20                | 66         | 69.3                 | 83.16          | 100                | 0.133725438  |
|                | Magnetorquer | 3.3         | 100               | 330        | 346.5                | 415.8          | 50                 | 0.334313595  |
| <b>OBC</b>     | OBC          | 5           | 40                | 200        | 210                  | 252            | 100                | 0.4052286    |
|                | Telemetry    | 5           | 300               | 1500       | 1575                 | 1890           | 11                 | 0.334313595  |
| <b>Rx-Tx</b>   | Beacon       | 5           | 20                | 100        | 105                  | 126            | 100                | 0.2026143    |
|                | GPS          | 3.3         | 40                | 132        | 138.6                | 166.32         | 30                 | 0.0802352628 |
| <b>Payload</b> | LoRa         | 5           | 20                | 100        | 105                  | 126            | 10                 | 0.02026143   |
|                |              |             |                   |            |                      |                |                    |              |
| <b>EPS</b>     | EPS          | -           | -                 | 160        | 168                  | 201.6          | 100                | 0.32418288   |
|                | Thermal      | -           | -                 | 250        | 262.5                | 315            | 32                 | 0.16209144   |
|                |              |             |                   |            | <b>Tot Power(mW)</b> | <b>3575.88</b> | <b>Tot. Energy</b> | <b>1.997</b> |

Table 4.2: Power Budget

Conventionally, EPS will work on different modes to manage the overall power production and distribution of the CubeSat. The main modes are initializing mode, which is during the initial phase of CubeSat launch and the normal mode, which

is the rest of the mission. The initializing mode is divided into three: **Pre- Launch mode, Launch mode and Initializing mode.**

In pre launch mode, all subsystems are off. In launch mode, that is when the CubeSat is deployed into orbit, the EPS and OBC turns on. Then, during initializing, all subsystems are turned on for a small amount of time to check whether all subsystems are working properly.

The normal mode consists of safe mode and nominal mode.

During safe mode, only the EPS and beacon of telemetry system works, and the CubeSat is in a power saving mode. The nominal mode is the general purpose mode where payload will function.

To calculate battery and solar panel specifications, the peak power budget of the CubeSat is only considered since the power requirements won't exceed that requirements. The other modes are only documented for designing the functioning of micro-controller which controls the EPS. Of all the modes, highest power consumption during sun phase is when transmission and payload active (1.016 Wh), and highest power consumption during eclipse phase is when transmission and payload active (0.769 Wh).

From the peak power budget table, the highest energy requirement is 2.307Wh.

# Chapter 5

## System Architecture

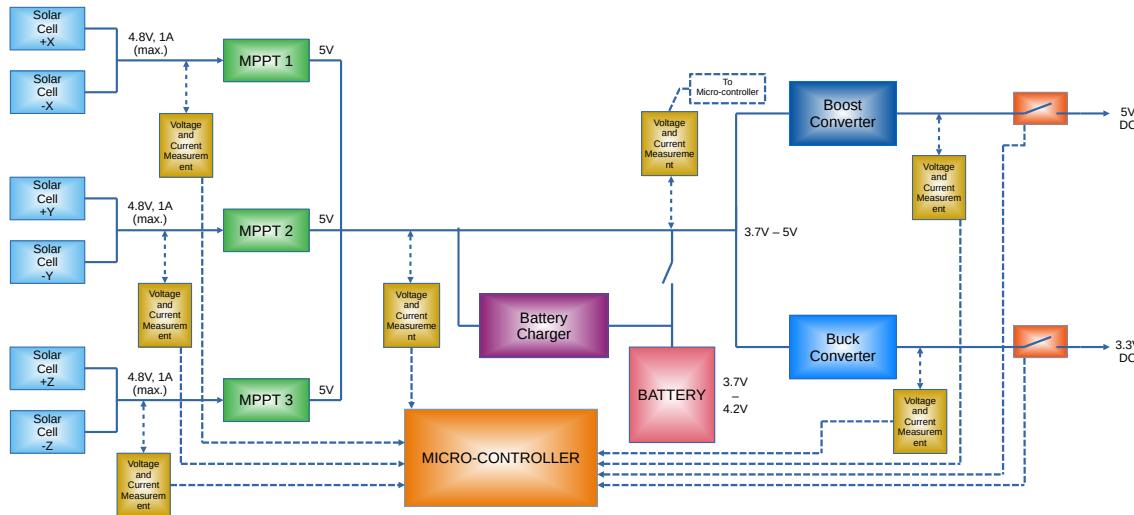


Figure 5.1: System Architecture of the CubeSat EPS

The electrical power system (EPS) is a critical component of CubeSat architecture, as it provides power to all the satellite's systems and payloads. EPS architecture of a CubeSat is designed to be highly efficient and reliable, while minimizing the mass and power consumption of the components. The EPS must be carefully designed and tested to ensure that it can withstand the harsh space environment and meet the mission requirements.

The CubeSat is equipped with six solar panels on each side. To ensure that the solar panels operate at their most efficient points, the solar panels are connected to

MPPT ICs such that the solar panels on the opposite sides are connected to single MPPT IC. Since only one of the opposite sides of the cube is irradiated at any time, the MPPT of two solar panels can be achieved using a single IC. This arrangement reduces the number of MPPT ICs required by the system. The MPPT runs a constant voltage algorithm and delivers 5V at the bus.

When the CubeSat is capable of generating enough power through the solar panels, the components are directly powered by the panels through the bus. At the same time, a battery charging IC uses some of the energy to recharge the Li-ion cell. The battery charger takes the 5V from the bus and converts it to the voltage and current optimal for battery charging depending on the state of charge, temperature of the cell etc. During eclipse the battery powers the energy from the panels is absent and the battery powers the CubeSat components. The voltage of the battery varies from 3.7V to 4.2V.

The buck and boost converters provide power to the 3.3V and 5V rails respectively. Their input voltage varies from 3.6V to 5V to account for direct power from the panels when sunlight is available and for battery power during eclipse.

To ensure proper functioning and to identify any faults in the system, there are various voltage and current measurement units placed at various points in the circuit. They measure the current and voltage at each point and the data is fed to a micro-controller, which is programmed to open the switches at the rails in case of any faults.

# Chapter 6

## Component Selection and Design

### 6.1 Solar Panels

TJ Solar Cell 3G30C - Advanced is selected. This cell is a GaInP/GaAs/Ge on Ge substrate triple junction solar cell. The end-of-life version of the 3G30C solar cell offers best EOL-performance values. Connected to the EPS via an external bypass diode protection.

Specifications:

- Average Open Circuit Voltage: 2.7V
- Maximum Power Point Voltage: 2.41V
- Average Short Circuit Current: 520.2 mA
- Maximum Power Point Current: 504.4mA

It has an average efficiency of 29.8% at  $1353 \text{ W/m}^2$ . This solar cell is excellent for space applications.

Since a CubeSat has six sides and one cell of selected model fits half of a side, a total of 12 cells can be placed.

Power from a solar panel, P :

$$P = P_i \times \text{Area} \times \eta \times \cos\alpha$$

Where,

$P_i$  = incident power from sun ( $1367 \text{ W/m}^2$  at Low Earth Orbit)

$\eta$  = conversion efficiency of cell (take 20% for contingency)

$\alpha$  = angle between normal of cell and incident light (avg:  $35^\circ$ )

Area of solar cell at one face :  $0.0032 * 2 = 0.0064 \text{ m}^2$

Therefore,  $P = 1367 \text{ W/m}^2 \times 0.0064 \times 20 \times \cos 35^\circ = 1.43 \text{ W}$

Assume two faces of CubeSat is lit by sunlight every time except eclipse and total sun time( $T_s$ ) is 1 hour (from orbital parameters).

Therefore, total energy produced is,  $P \times T_s = (1.43 \times 2) \times 1 = 2.86 \text{ Wh}$

**Of all modes specified in power budget, nominal mode at sun phase with transmission and payload ON consumes the most energy (1.016 Wh), and most energy consumed at eclipse phase is 0.769 Wh. This add upto 1.785 Wh, which is the worst case energy demand. This can be compensated by the designed solar panel configuration.**

Solar panels are connected in such a way that each side has two cells connected in series. The maximum voltage developed per side is 4.4V and the maximum current that can be generated per side at peak power point is 0.5A. Panels on opposite sides are connected in parallel.

## 6.2 Maximum Power Point Tracking Circuit

Maximum Power Point Tracking (MPPT) is a technique used in photovoltaic (PV) systems to optimize the power output of the PV panel. The goal of MPPT is to ensure that the PV panel operates at its maximum power point (MPP), which is the point on the current-voltage (I-V) curve of the panel where the panel can produce the maximum amount of power.

As each solar panel has different temperatures and incident radiance angles, the Maximum Power Point (MPP) is also different. So each solar panel has a MPPT

converter to assure that the maximum power available at the solar panels is transferred independently from their working power points. Since the peak power point cannot be accurately predicted, many different algorithms exist for finding the best approximation. MPPT controllers use different algorithms such as **Perturb and Observe (P&O)**, **Incremental Conductance**, and **Fuzzy Logic** to track the MPP. These algorithms operate by incrementally changing the load impedance of the panel and measuring the corresponding change in voltage and current until the MPP is reached.

The SPV1040 was chosen as the MPPT IC. It is a boost converter with duty ratio controlled by Perturb and Observe MPPT algorithm. The perturb and observe algorithm is based on monitoring either the voltage or the current supplied by the DC power source unit so that the PWM signal duty cycle is increased or decreased step-by-step according to the input power trend. This chip has inbuilt over-current protection and a cutoff mechanism if the solar panel connection is reverse-inserted to prevent damage to the IC and the external circuit.

The MPPT IC can only handle 1.8A of input current. Also, the input voltage of each MPPT must be much less than the desired output voltage for them to function properly. Depending on the configuration, the number of MPPT ICs was calculated to be 6.

Each side of cubesat consists of two 2.4V cells, each cells giving a maximum current output of 500mA for each MPPT, which is well below the maximum current the IC can handle. One cell from opposite sides are connected to its own MPPT, which is set to boost to 5.2V. The MPPTs are then connected in parallel.

Connecting multiple regulators in parallel can sometimes cause problems if one source is outputting a slightly smaller or larger voltage than the others. In most cases, the use of ballast resistors solves this problem. The footprint for a 25mohm ballast resistor will be chosen to be included in series with the output of each MPPT.

Specifications of SPV1040:

- Input Voltage: 0.3 - 5.5V
- Output Voltage: 5V
- Switching Frequency: 100kHz
- Efficiency: 95%

The MPPT circuit schematic is shown below:

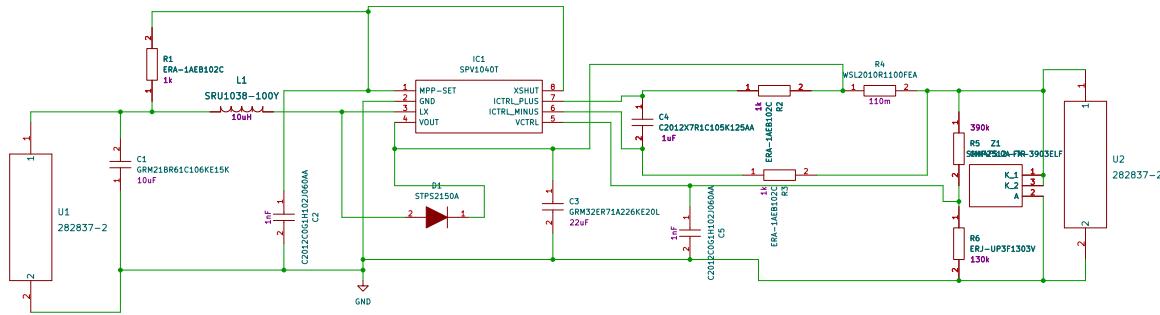


Figure 6.1: MPPT circuit with SPV1040

### 6.2.1 PCB Layout

The MPPT PCB is shown below:

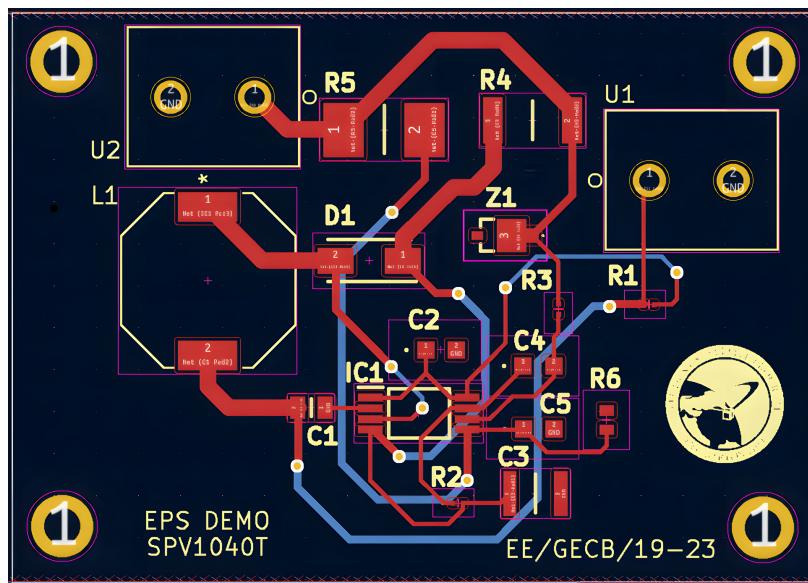


Figure 6.2: MPPT PCB Design

The MPPT 3D is shown below:

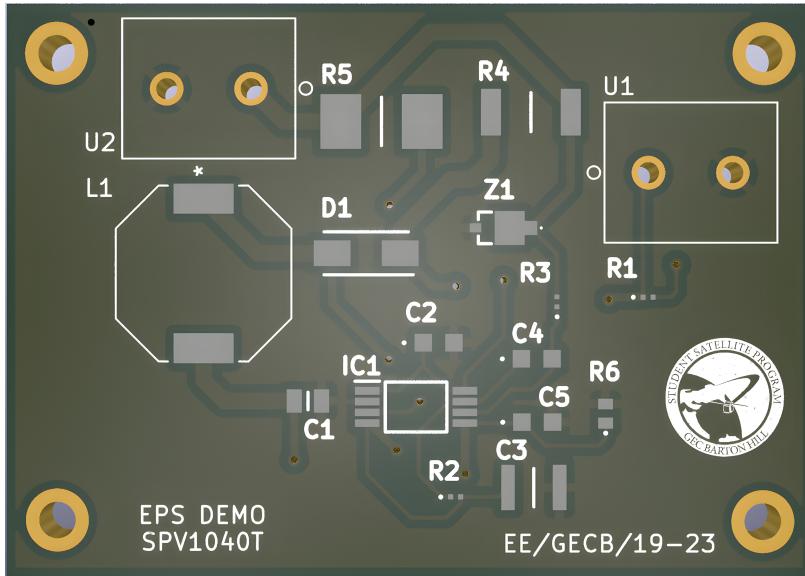


Figure 6.3: MPPT 3D

## 6.3 Battery

The most popular types of batteries use the following materials: Nickel Cadmium (NiCd), Nickel Metal Hydride (NiMH), Nickel Hydrogen (NiH<sub>2</sub>), Lithium Ion (Li-Ion) and Lithium Polymer (Li-Po). The Li-Po and Li-Ion became the standard use in space technology due to their high energy density (Upto 200 Wh / kg on Li-Po and upto 250 Wh / kg on Li-Ion) and also due to the number of charging cycles being as high as the NiMH, whilst presenting higher operating temperatures.

### 6.3.1 Battery Selection

According to solar power calculations, the solar panel can produce 2.86 Wh of energy in a single orbit, of which 1.016 Wh will be used up in sun phase in worst case ( $E_{max}$ ). So there will be atleast 1.844 Wh of energy left to charge the battery.

In power budget, the most amount of energy required during eclipse phase is 0.769 Wh.

Assign 50% contingency, the battery needs energy of 1.1535 Wh for charging, to

use in worst case eclipse mode. This energy corresponds to 30% of total battery capacity (End Of Life capacity of battery).

So, we need a battery with Beginning Of Life capacity of 3.845 Wh.

$$\text{Required Ah of battery} = \frac{\text{Estimated Wh}}{\text{Battery Voltage}}$$

$$\text{So, required Ah} = \frac{3.845\text{Wh}}{3.6V} = 1.068 \text{ Ah}$$

Considered options are Samsung 18650 and Panasonic 18650 Li-ion cells. Samsung 18650 has 2500 mAh capacity, 3.6V and a capacity loss of 60% after 250 charging cycles ( $N_{in}=250$ ). Panasonic 18650 has 3350 mAh capacity, 3.6V and a capacity loss of 60% after 300 charging cycles ( $N_{in}=300$ ).

for a CubeSat mission of 3 months, required battery cycles will be around 1460 cycles (since each day, CubeSat goes through 16 sun phase).

Total no. of cycles a cell can provide =  $N_T$

$$N_T = \frac{N_{in} \times C_B}{C_{exp}}$$

Where,

$N_{in}$  = initial number of cycles a battery can provide)

$C_B$  = Capacity of the battery

$C_{exp}$  = Capacity expected to use

$$C_{exp} = \frac{E_{max}}{V_B}$$

Where,

$E_{max}$  = Maximum energy consumed in a phase with 50% contingency

$V_B$  = Battery voltage

Therefore,  $C_{exp} = \frac{1.524}{3.6V} = 450 \text{ mAh}$

Total no. of cycles Samsung 18650 can provide =  $\frac{250 \times 2500}{450} = 1388$  cycles

Total no. of cycles Panasonic 18650 can provide =  $\frac{300 \times 3350}{450} = 2200$  cycles

**Therefore, Panasonic 18650 is selected, as it can compensate the charging cycle requirements.** Specifications of Panasonic NCR 18650 GA Li-Ion cell:

- Voltage: 3.7V - 4.2V
- Capacity: 3500mAh
- Can withstand 2200 charge-discharge cycles
- 1800 cycles till capacity reduces to 60%

## 6.4 Battery Charger

The battery also needs a charger to regulate its current and voltage while charging. BQ25302, a synchronous Buck Battery Charger IC was selected and connected in external power path mode as given in the data-sheet. The BQ25302 is a highly-integrated standalone switch-mode battery charger for single cell Li-Ion batteries. It supports input voltage in the range of 4.1V to 6.2V and maximum charging current of 2A. It has a very low quiescent current of 200nA to conserve battery during idle. It detects the charge voltage setting at startup and charges the battery in four phases: battery short, pre - conditioning, constant current, and constant voltage.

The charger incorporates various safety features for battery charging and system operations, including battery temperature monitoring based on negative temperature coefficient (NTC) thermistor, charge safety timer, input over-voltage and over-current protections, as well as battery over-voltage protection. Thermal regulation regulates charge current to limit die temperature during high power operation or high ambient temperature conditions.

The STAT pin output reports charging status and fault conditions. It is connected to REGN via a current limiting resistor and LED. Blinking LED indicated a fault condition.

The battery charge voltage setting is done by the VSET pin. As the Panasonic NCR 18650 GA Li-Ion cell with a charge voltage of 4.2V was selected the VSET pin was shorted to ground to set the battery charge voltage to 4.2V.

To preserve battery life, the charging current has to be limited. The charging current provided to the battery by BQ25302 IC can be limited by adjusting the resistor value at the ICHG pin according to the equation below:

$$I_{CHG}(A) = K_{ICHG}(A.\Omega)/R_{ICHG}(\Omega) \quad (6.1)$$

Where,

- $I_{CHG}$  is the charging current
- $K_{ICHG}$  is the charge current ratio. The  $K_{ICHG}$  vs.  $I_{CHG}$  typical characteristic curve is shown in fig. 6.4
- $R_{ICHG}$  is the resistor value at the ICHG pin

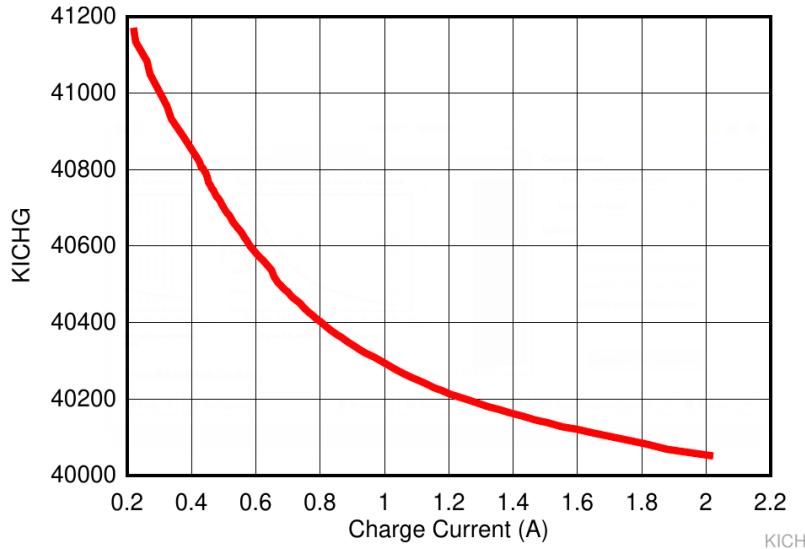


Figure 6.4:  $K_{ICHG}$  vs.  $I_{CHG}$  characteristics (Source: BQ25302 data-sheet)

From the fig. 6.4, it is clear that the value of  $K_{ICHG}$  should be 40200 to limit the charging current near to 1.2A. Thus, from equation 6.1 the calculated value of resistance at the ICHG pin is 33.5kΩ.

The IC is connected in an external power path configuration by connecting an external PFET for the battery. The PFET turns on when there is no external supply available from the solar panels and the system has to be powered by the battery. It is also useful if the system has to be powered ON when the battery is over-discharged or dead.

#### Specifications and Operating Conditions of BQ25302:

- Input Voltage: Upto 5V
- Output Voltage: Upto 4.2V
- Switching Frequency: 1.2MHz
- Output Current: Limited to 1.2A by connecting a 33.5k resistor at ICHG pin
- Efficiency: 94.3% at 1A from 5V input
- Thermistor: Semitec 103AT-2 (10kΩ)
- Charging Temperature: Limited between 0 - 45 °C

The Battery Charger circuit schematic is shown below:

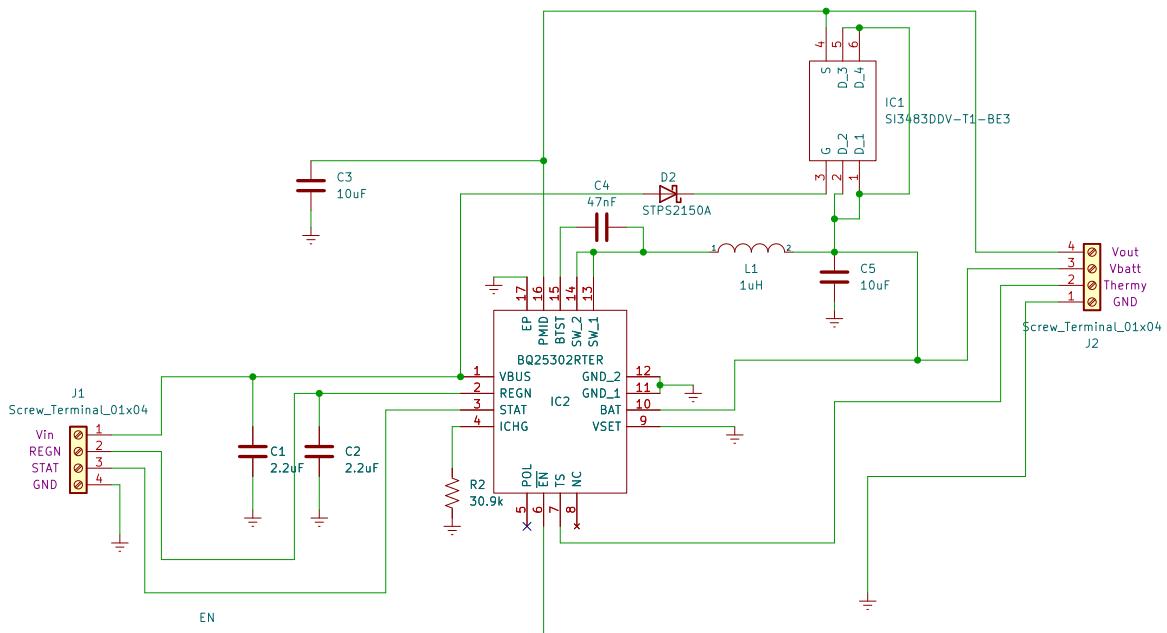


Figure 6.5: Schematic of Battery Charger circuit with BQ25302

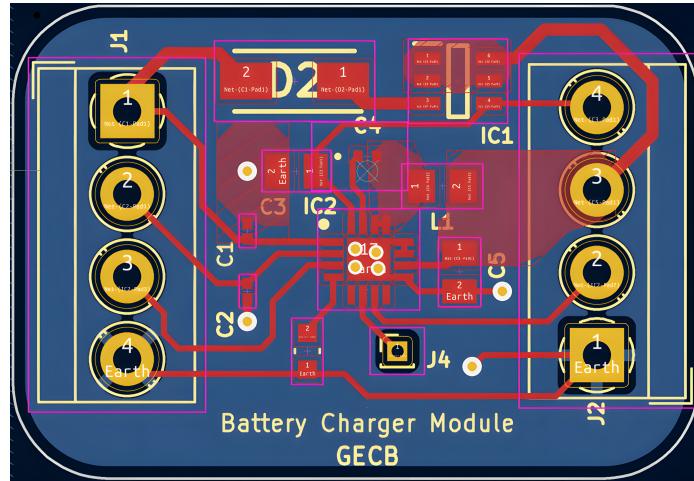


Figure 6.6: Battery Charger circuit PCB - All Copper Layers only

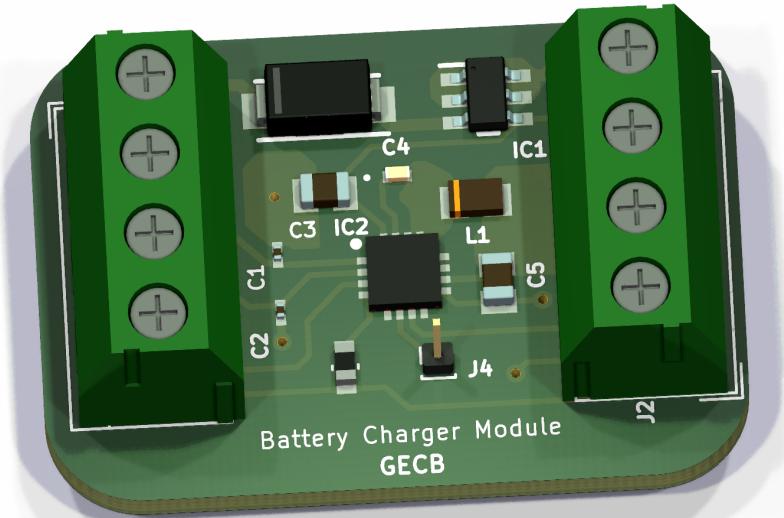


Figure 6.7: Battery Charger circuit PCB - 3D Model

## 6.5 Buck and Boost Converters

The power conditioning is associated with regulating the voltage to accommodate for the charging voltage and the voltages of the satellite's subsystems. In most subsystems, the need for a specific voltage requires a regulation of either a step-up or a step-down of the supplied voltage. It can be done by buck convertor(step-down) and boost converter(step-up).

TPS62203 was selected as the buck converter to provide step down voltage of the DC bus to supply the 3.3V loads. It is a high-efficiency synchronous step-down converter with up to 95% efficiency. It operates with a 1MHz fixed frequency (PWM) at moderate loads to heavy load currents and operates with pulse frequency modulation (PFM) at light load currents.

It has a shutdown quiescent current of typically  $0.1 \mu\text{A}$

Specifications and Operating Conditions of TPS62203:

- Input Voltage: 3.6 - 5V
- Output Voltage: 3.3V
- Switching Frequency: 1MHz
- Output Current: 300mA (max.)

LTC3426 was selected as the boost converter to provide step up voltage of the DC bus to supply the 5V loads. It has an internal 2A MOSFET as the switch.

Specifications and Operating Conditions of LTC3426:

- Input Voltage: 3.6 - 5V
- Output Voltage: 5V
- Switching Frequency: 1.2MHz
- Output Current: 500mA (max.)

All convertors operate in continuous conduction mode.

The Buck and Boost Converter circuit schematic is shown below:

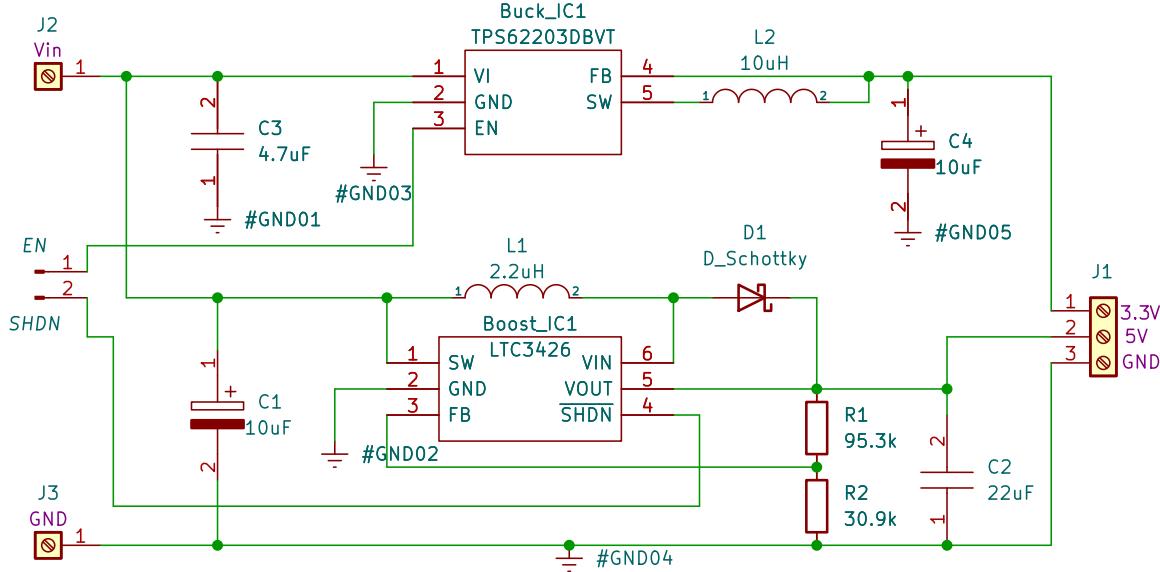


Figure 6.8: LTC3426 as boost IC and TPS62203 as buck IC

## 6.6 Protection and Measurement Circuits

LTC4361-2 is selected as the over voltage and over current protection IC. It controls an external N-channel MOSFET as a switch to cut the path of current if there is an event of over current or voltage. Manual control of the MOSFET is also possible which may be useful to turn off power to buses by the micro-controller as per different modes of operation of the CubeSat.

Voltage and current passing through each bus and subsystems are continuously monitored by sensors and this information is fed to the micro-controller. These measurements help in estimation of load requirement of subsystems and also help in triggering of protection circuits if a subsystem needs to be turned off in case of an occurrence of a fault.

LTC2990 was chosen as the voltage and current monitor. It can measure the voltage of four external channels and its supply voltage ( $V_{cc}$ ) simultaneously. It has a 14 bit ADC for measurement. The LTC2990 has the ability to perform 14-bit current measurements with the addition of a current sense resistor. The measurements are passed on to the micro-controller through the two wire I<sub>2</sub>C interface.

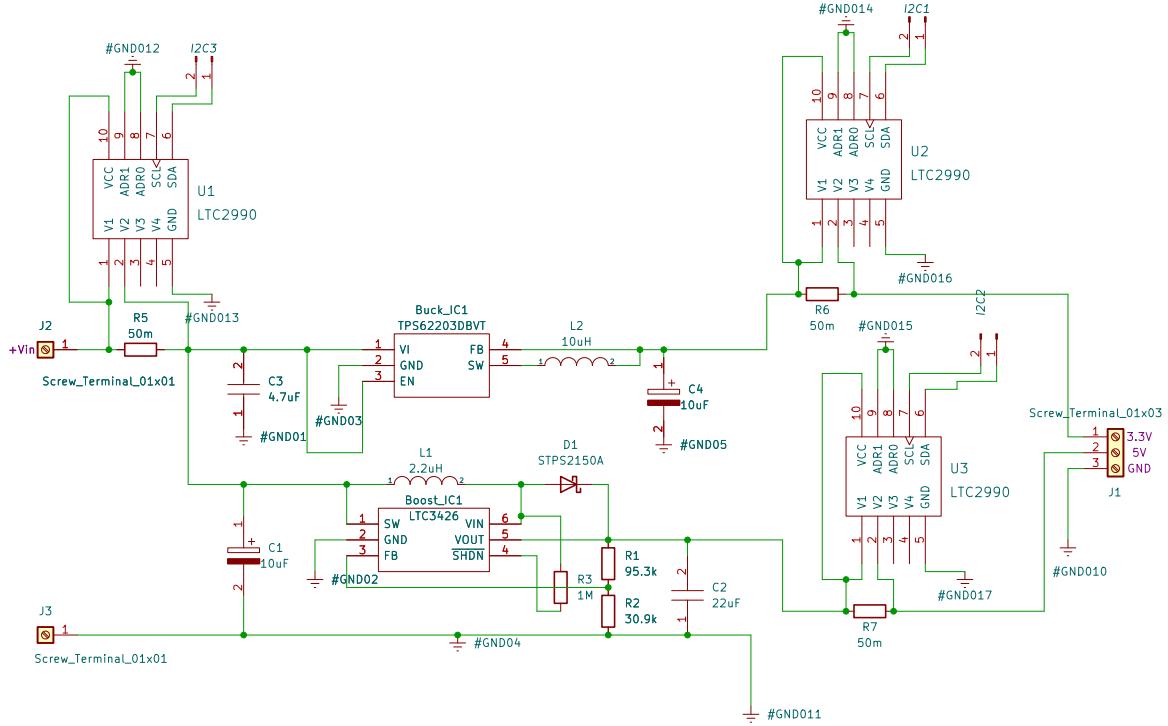


Figure 6.9: Schematic of Buck and Boost circuit with Protection and Voltage and Current Measurement ICs

The screw terminals are used to connect to the power inputs and outputs of the PCB. Pin header connectors are provided for connecting IC control pins to the micro-controller. The pin headers and their functions are given below:

- EN pin of Buck IC: Setting LOW turns off the buck IC
- SHDN pin of Boost IC: Setting LOW turns off the boost IC
- ON1 pin of IC1 (LTC4361): Setting HIGH turns off the MOSFET (IC2 - SI470DH) of the 3.3V bus
- ON2 pin of IC4 (LTC4361): Setting HIGH turns off the MOSFET (IC3 - SI470DH) of the 5V bus
- SCLSDA1 pins: I2C connection to micro-controller relaying measured parameters of 3.3V bus by U2 (LTC2990)

- SCLSDA2 pins: I2C connection to micro-controller relaying measured parameters of 5V bus by U3 (LTC2990)
- SCLSDA3 pins: I2C connection to micro-controller relaying measured parameters of input bus ( $V_{in}$ ) by U1 (LTC2990)

### 6.6.1 PCB Layout

A double layer PCB was designed from the schematic (Ref. Fig. 8.4). The second layer was used as the GROUND plane. The schematic and PCB design was done in KiCad EDA.

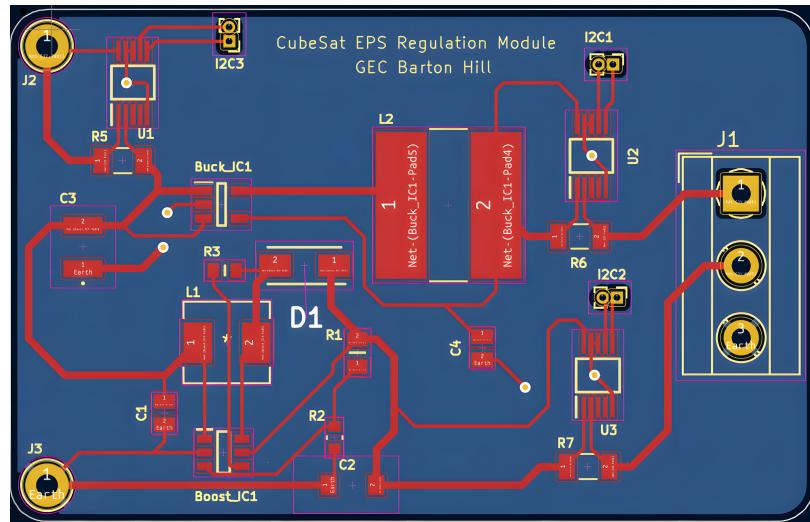


Figure 6.10: Buck and Boost circuit PCB - All Copper Layers only

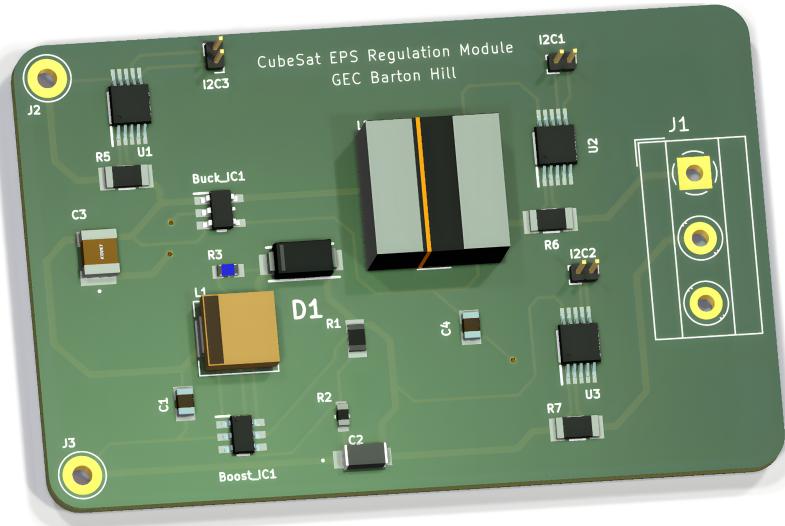


Figure 6.11: Buck and Boost circuit PCB - 3D Model

## 6.7 Micro-controller

The requirements of micro-controllers depend of the BUS, the other links as digital/analog outputs, the energy consumption, the voltage range and the memory.

The micro-controller has to handle the energy distribution of the entire Cube-Sat, and has to communicate with the OBC of CubeSat. Thus, it needs digital/analog outputs (for lines going to ADCS, Telemetry, OBC, battery , monitoring IC etc), UART for two-way communication between the micro-controller and the OBC and SPI to send data of the battery level to the micro-controller.

The micro-controller should have low energy consumption, around the micro ampere range for the “Active mode” and around the hundreds of nano ampere for the “Off mode”. It also should need sufficient memory storage to store the readings to provide it to OBC when requested. It should also independently take action during over-current, over voltage happenings, and also should control different power modes of the EPS.

Few considered options were Arduino, Raspberry Pi, TI micro-controllers, STM32 etc. Arduino has limited programming capabilities, and Raspberry Pi was too large for the need and also has high power consumption. STM32 was selected due to its wide range of capabilities, still maintaining the size and power constraints. It also has more community support which makes the programming easier.

STM32 is a ARM Cortex M4 32-bit micro-controller having a flash memory of 512 Kb. It can have upto 81 I/O ports with interrupt capabilities. It can have upto 78 fast I/Os upto 42 MHz, 3 I2C interfaces, 3 USARTs, 4 SPIs etc. We selected STM32401RE development board to test its capabilities, and one of its advantage is that, it can be programmed using Arduino IDE or even MATLAB, apart from its own programming IDE called STM32CUBE IDE.

## 6.8 List of Components

| Sl. No. | Component              | Description        |
|---------|------------------------|--------------------|
| 1       | TPS62203               | Buck Converter IC  |
| 2       | LTC3426                | Boost Converter IC |
| 3       | BQ25302                | Battery Charger IC |
| 4       | Panasonic NCR 18650 GA | Battery            |
| 5       | SPV1040T               | MPPT               |
| 6       | TJ Solar Cell 3G30C    | Solar Cell         |
| 7       | STM 32 F401RE          | Micro-controller   |
| 8       | Resistors              |                    |
| 9       | Capacitors             |                    |
| 10      | Inductors              |                    |
| 11      | PCB                    |                    |

Table 6.1: Components Required

# **Chapter 7**

## **Component List**

# **Chapter 8**

## **Conclusion**

### **8.1 Conclusion**

After conducting proper literature study, we have decided upon the type of architecture to be implemented on the EPS and have formed the specifications of the components. The design of all the circuits was also completed and PCB design was started. Next, we are planning to procure all the components and design the PCB layouts. We look forward to completing the project as per the schedule.

### **8.2 Future Scope**

The STM 32 Micro-controller could be used to monitor and display the voltage and current levels of the different buses and hence determine whether a bus is operating or not. Also, the micro-controller could be used to provide the PWM for the converter in the MPPT circuit depending upon the available irradiance by following an algorithm.

# **Chapter 9**

## **Testing**

### **9.1 Variable DC load using MOSFET**

To test the efficiency of the demo boards, different tests like line regulation, load regulation etc have to be conducted. For line regulation, at a certain constant load, we should create a voltage values table and get the efficiency curve. For that, we need different values of resistance (load). Instead of using resistance, we could use a MOSFET for simulating various load conditions. Since we doesn't need any higher loads to test the demo boards, a circuit that simulate a load precisely for values from 0 to 2A and with a precision of 0.01A will do the job.

The working of the circuit is simple. The MOSFET is controlled by voltage on its gate. Till the voltage between gate and source ( $V_{gs}$ ) is not higher than a certain value called  $V_{th}$ , the MOSFET is disabled, so no current will be flowing between drain and source. After  $V_{th}$ , the MOSFET starts a current flow and the value of this current will get higher as  $V_{gs}$  increases till a certain value where the MOSFET is fully on. So, by varying the voltage at the gate we change the current flow and by that we can simulate the load value.

But there is no relation between the voltage at the gate and the current value. For that we add a 1 ohms 5W resistor at the source of the MOSFET. Since we know for sure the resistor is 1 ohm, for example, when 1 amp is flowing, we will have 1V

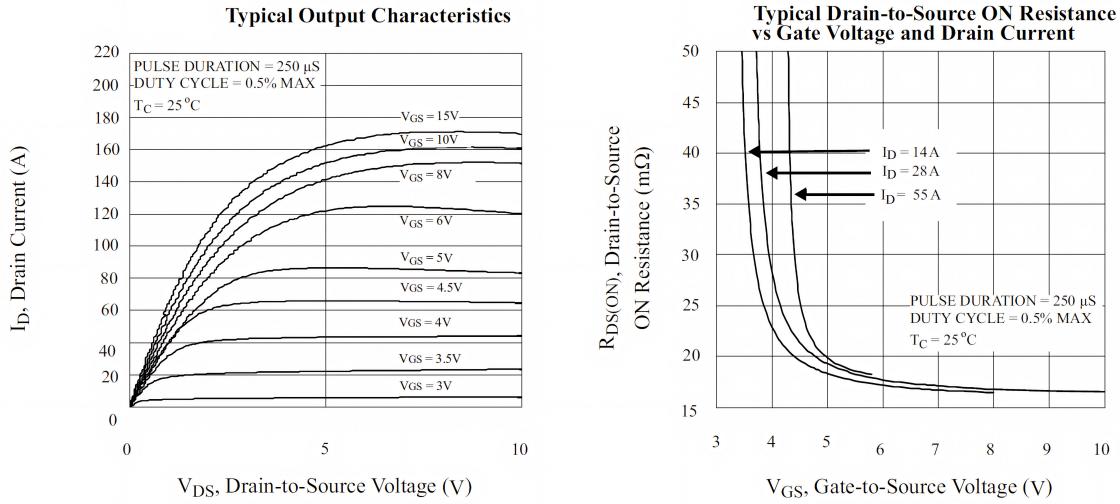


Figure 9.1: Mosfet characteristics

voltage drop on this resistor and we can measure that with a precise ADC. In this way we can know the current value.

Since we can't know what voltage to put at the MOSFET gate in order to get any current value, we use the OPAMP and this will do that for us automatically. Let's say we want a current flow of 1A. All we need is to get a 1V voltage drop on the 1ohm power resistor. That point, as you can see in the schematic, is connected to the negative input of the OPAMP. We know that the OPAMP will do all that is necessary to have the same voltage on the input V- as on input V+. So if we place 1V at V+, the OPAMP will change the voltage at the MOSFET gate till the voltage on the resistor is 1V as well, since that point is connected at input V- of the OPAMP. So, just like that, we can get any value of current that we want.

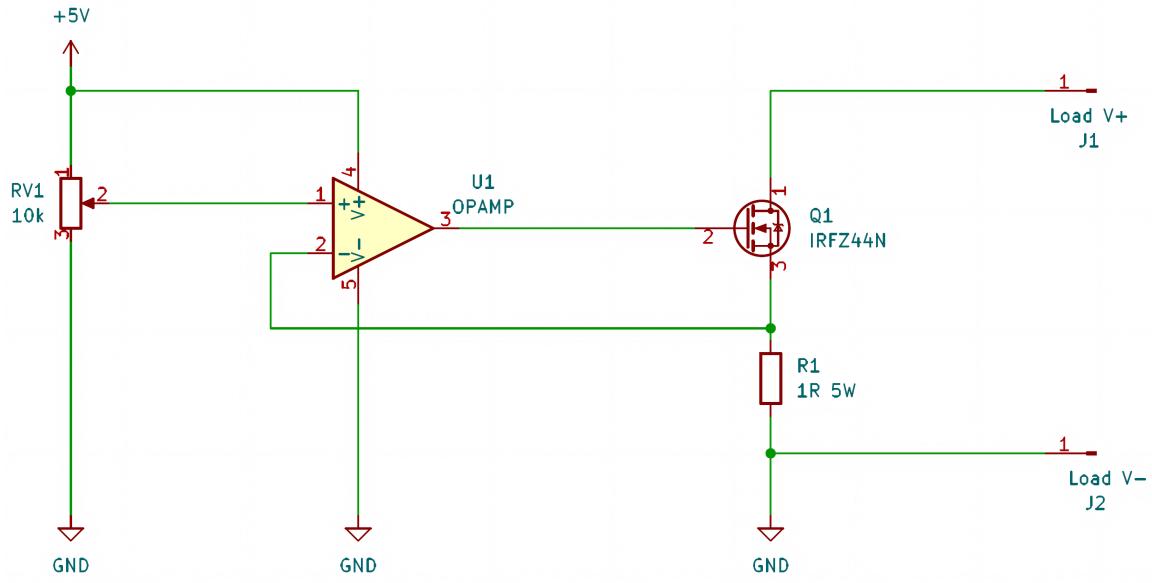


Figure 9.2: Variable DC load schematics

## 9.2 Testing of DC/DC converters

DC/DC converters have a specified input voltage operating range. To confirm the DC/DC converter works properly over the entire range of input voltages, they are tested using an adjustable or programmable dc source to provide the input voltage. A dc electronic load is used on the output of the DC/DC converter to set the output load current and simulate the device that the DC/DC converter would power.

### 9.2.1 Input turn-on, input turn-off voltage levels and timing test

To test the minimum input voltage turn-on level, the DC/DC converter is turned on using the nominal input voltage while using the electronic load to apply the maximum rated output current or power. The input voltage is then reduced until the unit output begins to drop or the minimum input voltage setting is met.

To confirm the DC/DC converter would turn on with a maximum load on the output, the input voltage would be set to the minimum and toggled off and back

on while measuring the output voltage and current. The output voltage and ripple and noise may also be measured to see if the lower input voltage setting has any effect on the output stability or ripple.

Additionally, this setup can be used to test and measure the turn-on time, turn-off time, and hold-up times. An oscilloscope is used for the ripple and for periodic and random deviation (PARD) measurements.

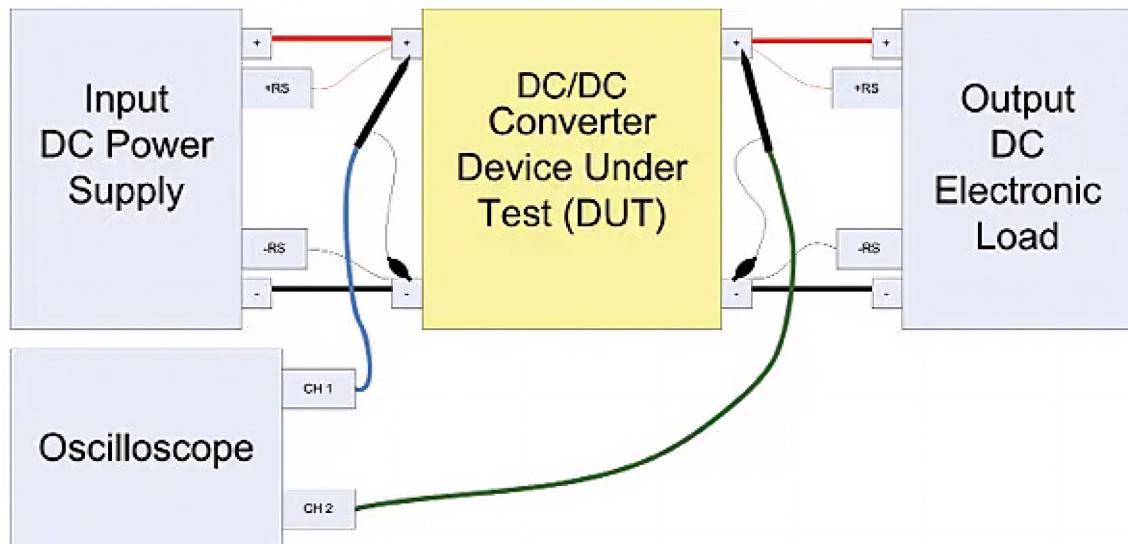


Figure 9.3: Test setup

Turn-on time indicates the period from the time at which the minimum input voltage is applied to the time the output voltage is within the output regulation limits. Turn-off time indicates when the input voltage drops below the specified minimum and the output turns off or drops to zero volts.

The hold-up timing test can use the same setup as the input turn-on and turn-off test. Hold-up timing indicates the time from when the input drops below the

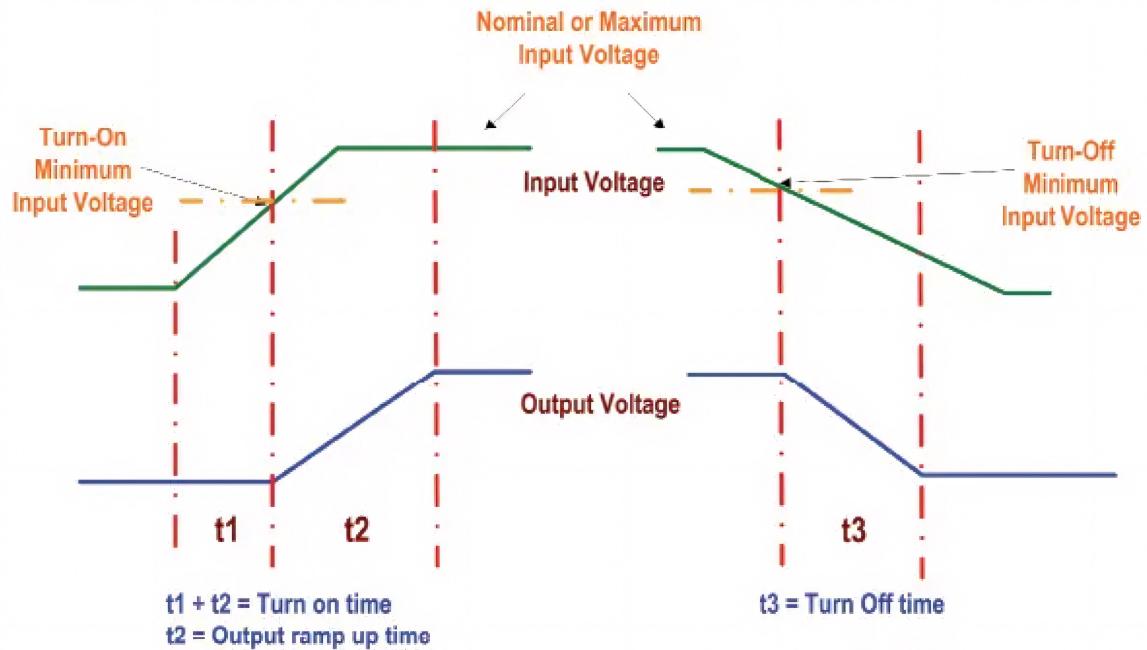


Figure 9.4: Turn-on, Turn-off characteristics

minimum input voltage and the output voltage drops below its minimum regulated output tolerance. This test also indicates how well the output of the DC/DC converter can continue to operate despite short interrupts and drops in the input voltage. Some DC/DC converters have an input fault detection signal. If so, this signal can be used to trigger the test.

### 9.2.2 Output line regulation

This test confirms that the output voltage stays within specified regulation limits when the input voltage varies from minimum to maximum operating voltage, as defined in the DC/DC converter specification. During this test the output load is usually set to nominal or maximum current as specified.

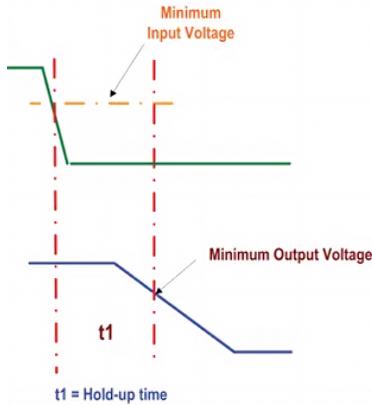


Figure 9.5: Hold-up time characteristics

The output line regulation test involves monitoring the output voltage and recording the total voltage deviation while varying the input voltage from its minimum to maximum specified limits. Some specifications show the output tolerance as a voltage (i.e. 3.3 Vdc  $\pm$  0.02 V) or as a percentage (i.e. 3.3 V  $\pm$  0.5%).

Output regulation  $R_o$  is calculated as a percentage and given by the equation:

$$R_o = \left| \frac{V_{omax} - V_{omin}}{V_{onoma}} \right| \times 100$$

where:  $V_{omax}$  = Vout at Vin max;  $V_{omin}$  = Vout at Vin min; and  $V_{onoma}$  = Vout nominal.

### 9.2.3 Output load regulation

The output load regulation test ensures the DC/DC converter output voltage stays within the specified regulation tolerance. Here, the change in output voltage is recorded while load is varied from minimum to maximum current. This delta voltage is used to calculate the percentage of deviation which is compared to the specified load regulation limits. Load regulation  $L_r$  is calculated as a percentage from the equation:

$$L_r = \left| \frac{V_{oio} - V_{oim}}{V_{onom}} \right| \times 100$$

where:  $V_{oio}$  = Vout at Iout max;  $V_{oim}$  = Vout at Iout min; and  $V_{onom}$  = Vout nominal.

#### **9.2.4 Output transient response deviation and time**

This test determines how well the output voltage responds to a sudden change in output current. The measurement includes the maximum output voltage deviation and the time it takes for the voltage to recover to its regulated output nominal voltage tolerance.

For this test, a dc electronic load is set for a minimum and maximum current and then set for the slew rate for the rising and falling edge of the current transition. The frequency and duty cycle can also be set for the pulsed current.

#### **9.2.5 Output ripple and noise voltage**

The output ripple and PARD reflects the output voltage of the DC/DC converter and its ability to filter out ripple and noise. Various topologies used in DC/DC converters have different internal switching frequencies which are reflected in the output ripple frequency. For example, internal chopper circuit transients can generate higher frequency noise. The output noise and ripple are measured using an oscilloscope. To avoid erroneous noise, it is important to minimize the length of the ground wire on the voltage probe.

#### **9.2.6 Output over-current protection**

The output over-current protection is intended to protect the DC/DC converter and the device it powers when the load exceeds the converter's maximum rated

current. There are different methods used in over-current protection. But the typical approaches are fold-back current limit and pulsing current-limit. The latter is generally referred to as hiccup-mode current limit.

Here are the differences between the two methods: In fold-back current limiting, the output voltage begins to drop and limits the output current supplied to the load as the load current rises above the current-limit set point. In hiccup current limiting, the output turns off when the output current exceeds the rated current limit point. It eventually turns back on. If the load continues to exceed the current-limit set-point, the output will continue turning on and off, hence the hiccup-mode name.

### 9.2.7 Output over-voltage protection

Most DC/DC converters have a built-in protection circuit that will shut off the output of the device when the output voltage is detected to be over the maximum limit. This facility is referred to as over-voltage protection (OVP). This protects the DC/DC converter from external excessive voltage applied to the converter output. If the DC/DC converter has an adjustable output (trimmed or programmable output voltage), it may be possible to increase the output voltage until the OVP point is exceeded and the protection circuit activates.

If the DC/DC converter does not have an adjustable output, an external voltage source can be applied across the output terminal, increased to the OVP trip point, then removed to see if the output has triggered and turned off.

DC/DC converters having an OVP fault signal can use it to determine if the output detected the OVP and, if so, shut off the output. The output voltage is monitored to determine when the OVP happened and then compared to the OVP specified limits.

### **9.2.8 Output operating temperature and over-temperature protection**

DC/DC converters have an operating temperature range, and many have an over-temperature protection (OTP) circuit that will turn off the output if temperature gets sufficiently high. For this test, a thermal chamber can raise and lower the DC/DC converter temperature to simulate the operating temperature range.

Thermocouples and thermal probes or infrared thermal measurement devices can measure the temperatures on the body of the DC/DC converter. During the operating temperature test, the DC/DC converter is loaded to its maximum rating for current and power. Meanwhile, the output voltage is monitored to verify it stays within specified limits.

During the test, the device temperature is recorded while monitoring the output voltage until the OTP circuit triggers and the output shuts off. The unit is then allowed to cool and input voltage is toggled off and on to verify the DC/DC converter recovers from the OTP.

### **9.2.9 Efficiency**

Efficiency determines the internal power dissipated by the DC/DC converter and how efficiently input power transfers to the converter output. This test usually takes place at the nominal input voltage and with the output load set to nominal or maximum specified ratings. The input voltage, current, and power is measured while the same parameters are measured on the output. Efficiency percentage  $E_p$  comes from the equation:

$$E_p = \left| \frac{V_{out} - I_{out}}{V_{in} - I_{in}} \right| \times 100$$

where: Vout and Iout = converter output voltage and current; Vin and Iin = converter input voltage and current.

This test can also capture the efficiency at various power levels. It's common to plot the data to show efficiency versus output-current.

# Chapter 10

## Appendix

### 10.1 Calculations

#### 10.1.1 SPV1040

Component Values These are constraints or known values for the calculations  $F_{sw} = 70\text{kHz} - 130\text{kHz}$  : Typ.  $100\text{kHz}$

$$V_{IN\_rp\_max} = 1\text{V}$$

$$I_{SC} = 0.52\text{A}$$
 (short circuit current)

$$I_{MP} = 0.504\text{A}$$
 (current at MPP)

$$V_{SC} = 2.7\text{V}$$
 (short circuit voltage)

$$V_{MP} = 2.41\text{V}$$
 (voltage at MPP)

$$T_{MPP} = 1\text{ms}$$
 (tracking time)

$$R_{MPPTset} = 1k\Omega$$

$$V_{OUT\_rp\_max} = 0.01\text{V}$$
 (maximum output voltage ripple)

$$V_{OUT\_max} = 5.2\text{V}$$
 (desired maximum output voltage)

$$I_{OUT\_max} = 1.8\text{A}$$
 (limit of spv1040 IC)

#### INPUT CAPACITOR

$$C_{IN} \geq \frac{I_{SC}}{F_{SW} * V_{IN\_rp\_max}}$$

$$= \frac{1\text{A}}{100\text{kHz} * 1\text{V}}$$

$$= 10\mu\text{ F}$$

#### INPUT VOLTAGE SENSING CAPACITOR

$$C_{MPPTset} \leq T_{MPP} * \frac{1}{R_{MPPTset}}$$

to be continued....

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