

*** Q-1, DYNAMICS OF GUIDED PROJECTILE ***

INITIAL CONDITIONS ASSUMED

Bomb cell is launched at 3000 m, with velocity of 20 m/s at different launch angles 0, 5 & 10 degrees.

Mass of Bomb cell is 160 kg.

MATLAB code

```
>> MAIN_Q1.m
```

```
%% MAIN_Q1
```

```
close all
```

```
clear all
```

```
clc
```

```
v0 = 20; % m/s
```

```
x0 = 0; % m
```

```
y0 = 3000; % m
```

```
launch_all = [0 , 5, 10];
```

```
for i = 1:3
```

```
    launch = launch_all(i);
```

```
    gamma_0 = deg2rad(launch); % rad
```

```
states0 = [v0, gamma_0, x0, y0];
```

```
[t,y] = RK4(@pointMassModel, 50, 0.01, states0);
```

```
figure(i);
```

```
subplot(4,1,1);
```

```
plot(t,y(:,3),'r');
```

```
% xlim([0 25])
```

```
ylabel('x');
```

```
grid on
```

```
subplot(4,1,2);
```

```
plot(t,y(:,4),'r');
```

```
% xlim([0 25])
```

```
ylabel('y');
```

```
grid on
```

```

subplot(4,1,3);
plot(t,y(:,1),'r');
% xlim([0 25])
ylabel('v');
grid on

subplot(4,1,4);
plot(t,y(:,2),'r');
% xlim([0 25])
ylabel('\gamma');
xlabel('t');
grid on

% figure(i);
%
% if i==1
%     title("\gamma_0 = 0^\circ");
% else if i==2
%     title("\gamma_0 = 5^\circ");
% else
%     title("\gamma_0 = 10^\circ");
% end

end

```

>> DENSITY.m

```

function rho = DENSITY(hg)

hg1=0;           % Gradient layer 1-2 (0 to 11 km)
T1=288.16;
a1=-0.0065;

p1=101325; % in pascal at MSL (Mean Sea Level)
d1=1.225; % in kg/m3 at MSL
g0=9.81; % gravitational acceleration at MSL
R=287; % in J/kgK

r=6371000; % Radius of earth

h=(r*hg)/(r+hg); % Calculation for Geopotential altitude
h1=(r*hg1)/(r+hg1); % Geo potential ALT CALC
T=T1+(a1*delh); % TEMP CALC
d=d1*((T/T1)^((-g0/(a1*R))-1)); % DENS CALC

rho = d;

end

```

>> POINTMASSMODEL.m

%% POINT MASS MODEL

function dstates = pointMassModel(~,states)

v = states(1);

gamma = states(2);

x = states(3);

y = states(4);

rho = DENSITY(y);

S = 0; **% m^2 (Assuming Point mass)**

cd0 = 0.302;

m = 160; **% kg**

g = 9.81; **% m/s**

D = 0.5*rho*v^2*S*cd0;

v_dot = -(D + m*g*sin(gamma))/m;

gamma_dot = -g*cos(gamma)/v;

x_dot = v*cos(gamma);

y_dot = v*sin(gamma);

dstates = [v_dot; gamma_dot; x_dot; y_dot];

end

>> RK4.m

%% RK4

function [t,y] = RK4(dydt, tf, h, y0)

% reshape y0 to column vector

y0 = reshape(y0, [], 1);

n = tf/h;

d = size(y0,1);

t = zeros(n,1);

y = zeros(n,d);

tn = 0;

yn = y0;

t(1,:) = tn;

y(1,:) = yn';

```

for i=2:n+1
k1 = dydt(tn, yn);
k2 = dydt(tn + h/2, yn + h*k1/2);
k3 = dydt(tn + h/2, yn + h*k2/2);
k4 = dydt(tn + h, yn + h*k3);

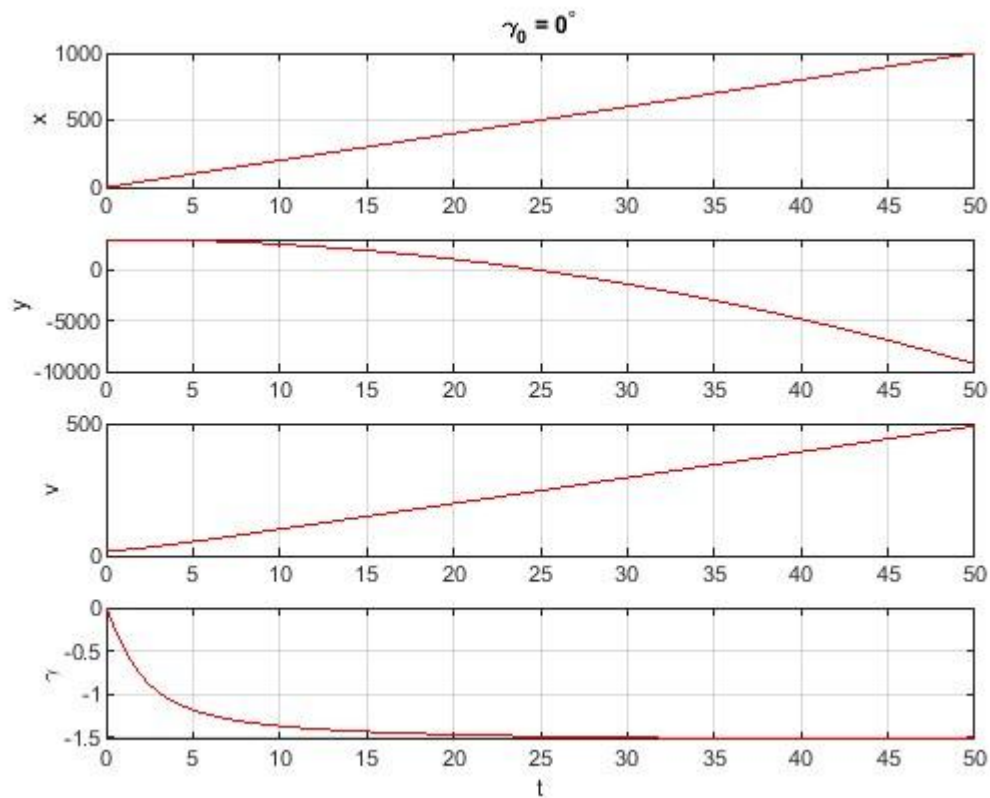
yn = yn + (k1 + 2*k2 + 2*k3 + k4)*h/6;
tn = tn + h;

t(i) = tn;
y(i,:) = yn';
end
end

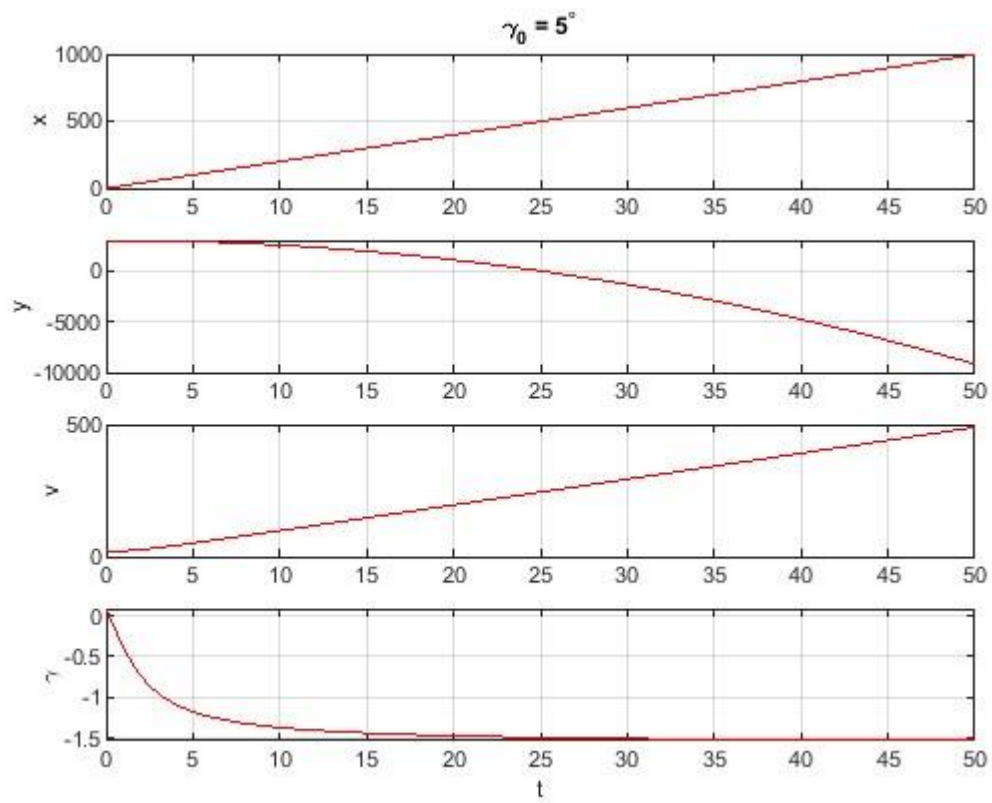
```

Output (PLOTS)

>> Launch angle = 0°



>> Launch angle - 5°



>> Launch angle - 10°

