
Short-term orbital effects of radiation pressure on the Lunar Reconnaissance Orbiter

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Abstract

Precision orbit determination for geodetic applications requires force models even for small perturbations. Radiation from the Sun and Moon is a significant source of perturbation in lunar orbits and inadequate modeling of radiation pressure (RP) can lead to large position errors. In this paper, we describe the short-term effect of RP on the Lunar Reconnaissance Orbiter (LRO), which has a position knowledge requirement of 50 m to 100 m in total and below 1 m radially. Models of varying complexity are compared to determine the computational cost of high-accuracy RP modeling. We find that the accelerations differ greatly depending on the Sun position, that a constant-albedo model is sufficient for lunar radiation, and that only a paneled spacecraft model can account properly for changing orientation and geometry of LRO. A spherical harmonics model for lunar albedo increases computational cost with little impact on the attained accuracy. If RP is neglected, the along-track position errors can be as large as 1100 m and the radial error varies periodically with an amplitude of up to 24 m/s, both of which violate the requirements.

Keywords

Radiation pressure modeling, Lunar Reconnaissance Orbiter, precision orbit determination

Acronyms: BRDF bidirectional reflectance distribution function; DLAM-1 Delft Lunar Albedo Model 1; LRO Lunar Reconnaissance Orbiter; RMSE root mean square error; RP radiation pressure; rRMSE relative root mean square error; Tudat TU Delft Astrodynamics Toolbox

1 Introduction

Precision orbit determination is cornerstone of satellite navigation and spaceborne geodesy. Only if the state, and particularly the position, of the spacecraft are known accurately can the high precision of modern instruments for gravity field recovery or satellite altimetry be exploited fully. Next to tracking data, force models have the largest role in improving orbit determination, accounting for gravity, solid tides, drag, and other accelerations. Another important non-conservative force is radiation pressure (RP), which can have magnitudes similar to third-body and irregular gravity field perturbations [1]. RP arises from the exchange of momentum between electromagnetic radiation and the spacecraft. Neglecting or mismodeling RP accelerations can deteriorate position knowledge below acceptable levels.

The Lunar Reconnaissance Orbiter (LRO) was launched in June 2009 to prepare for human missions to the Moon by identifying safe landing sites, locating resources, and characterizing the radiation environment [2]. To fulfil these objectives, LRO was equipped with instruments to, among other objectives, create high-resolution maps of the lunar topography and gravity field. Accuracies of 50 m to 100 m in the total position and sub-meter accuracy in the radial component are required to take advantage of the instrument resolutions [3, 4], which necessitates force models even for small perturbations. Solar RP “is the largest non-gravitational

perturbation affecting the LRO orbit and inadequate modeling [...] is the primary cause of large prediction errors for LRO, particularly during high-beta angle periods” [5]. The Moon itself is also a significant radiation source since there is no atmosphere and especially the lunar highlands are reflective [6]. The orbit determination error is also highly dependent on the modeling of how RP translates to accelerations; particularly during full-Sun periods, a model accounting for LRO’s geometry and the real orientation of the solar array and high gain antenna outperforms a simple spherical model [7].

This paper describes the short-term effects of RP on LRO’s orbit and their sensitivities to models of varying complexity. Other authors have described their orbit determination approaches for LRO [5, 7–13] but none compared RP modeling choices and their implications. Vielberg and Kusche investigated the effect of different models for Earth, where a plethora of observations are available and the radiation environment differs greatly from the Moon. The results of this paper can guide the choice of force models for orbit determination both in terms of accuracy and computational performance. Note that our results relate to short-term effects over 2.5 days, which is a typical arc used in orbit determination. Long-term effects, which may cancel or compound over the span of months, are not considered here.

The TU Delft Astrodynamics Toolbox (Tudat) was used for all orbital simulations and the models presented here were integrated into the software, which is freely available at <https://docs.tudat.space/>.

2 Radiation pressure modeling

2.1 Mechanics of radiation pressure

RP results from the momentum transfer between electromagnetic radiation and a surface. A spacecraft may receive such radiation from the Sun but also from other celestial bodies: planets and moons emit albedo radiation through reflection of sunlight and thermal radiation depending on surface temperature. The RP exerts a force on the spacecraft governed by surface properties such as area, reflectivity and absorptivity. The resulting acceleration is the result of a complex interplay of the bodies emitting radiation (the “sources”) and the spacecraft receiving the radiation (the “target”).

Radiation can be characterized by the radiant flux density, which commonly has units of W/m^2 . Radiosity is the *emitted and reflected* radiant flux density of an opaque surface. The irradiance E is the *incident* radiant flux density on a surface and provides a convenient way to decouple source and target models: the irradiance and the direction of incidence are sufficient to determine the target acceleration, independent of the actual source. We can combine this information into a vector quantity which we call directional irradiance $\mathbf{E} = E\hat{\mathbf{r}}_{t/s}$, where $\hat{\mathbf{r}}_{t/s}$ is the unit vector in the source-to-target direction. One or more directional irradiances, which can be thought of as light rays, are the output of a source model and used as input to the target model. The RP exerted on an irradiated surface is proportional to $1/c$, where $c = 299\,792\,458 \text{ m/s}$ is the speed of light. Given the magnitude of c , RP is usually small (around $4.5 \times 10^{-6} \text{ N/m}^2$ for solar radiation at Earth, where $E = 1361 \text{ W/m}^2$ [15]).

Electromagnetic radiation is often composed not just of a single wavelength but rather a range of wavelengths. The distribution can be described by the spectral irradiance in units of $\text{W}/(\text{m}^2 \text{ Hz})$. Since surface properties are often wavelength-dependent, the target model would also have to be aware of the distribution. However, the surface properties as a function of wavelength are often not known, which is also the case for LRO. Therefore, we assume the irradiance from source models to be integrated over the whole spectrum and the surface properties of the target model to be valid for all wavelengths.

2.2 Reflectance distribution

Describing the reflectance of a surface is key to RP modeling. Both the way a source reflects sunlight and the direction a target is accelerated depend on the angular distribution of reflectance.

General reflectance distribution In general, reflectance comprises a diffuse (scattered in many directions) and a specular (mirror-like) component. The remaining energy is absorbed by the surface. The reflectance varies with surface normal \mathbf{N} , incoming radiation direction \mathbf{L} , and observer direction \mathbf{V} . This geometry is shown in Figure 1. A bidirectional reflectance distribution function (BRDF) describes the fraction of irradiance reflected towards the observer per steradian, i.e. [16]

$$f_r(\theta_i, \phi_i, \theta_r, \phi_r) = \frac{dL_r(\theta_r, \phi_r)}{dE_j(\theta_i, \phi_i)}, \quad (1)$$

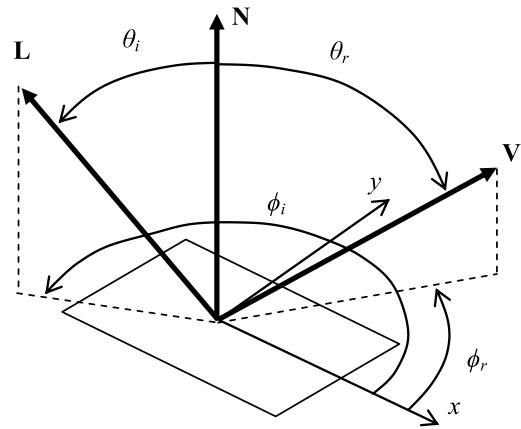


Figure 1. Geometry of a BRDF for a surface with normal \mathbf{N} , incoming direction \mathbf{L} , and observer direction \mathbf{V} . The viewing angle θ_r is between \mathbf{N} and \mathbf{V} . The phase angle (not labeled) is between \mathbf{L} and \mathbf{V} . Adapted from [16].

where dL_r is the reflected radiance (the directional counterpart to radiosity, typically in $\text{W}/(\text{m}^2 \text{ sr})$) and dE_j is the received irradiance.

The planetary surface BRDF directly leads to the albedo irradiance received by a target if the sun irradiance at the planet surface and the solid angle subtended by the target are known.

The target surface BRDF gives the direction in which the target is accelerated through integration over all directions \mathbf{V} in which radiation is reflected. The unitless reaction vector, which includes both the direction and magnitude based on absorbed, specularly and diffusely reflected fractions, is therefore [16]

$$\mathbf{R} = - \left[\mathbf{L} + \int_0^{2\pi} \int_0^{\pi/2} f_r \cos \theta_r \mathbf{V} d\theta_r d\phi_r \right]. \quad (2)$$

This vector encapsulates the mechanics of momentum transfer. The reaction is minimal for pure absorption ($f_r = 0$). The reaction is maximal (double the minimum) for pure specular reflection in the incidence direction.

Specular-diffuse reflectance distribution A simplified BRDF is usually more practical for RP modeling: the reflectance is assumed to be a mix of an ideal Lambertian diffuse component and a purely mirror-like specular component. Such a BRDF is given by [16]

$$f_r = C_d \frac{1}{\pi} + C_s \frac{\delta(\mathbf{V} - \mathbf{M})}{\cos \theta_i} \quad (3)$$

where C_d and C_s are the diffuse and specular reflectivity coefficients. Together with the absorption coefficient C_a , energy is conserved when $C_a + C_d + C_s = 1$. The vector $\mathbf{M} = 2 \cos \theta_i \mathbf{N} - \mathbf{L}$ is the direction of \mathbf{L} 's mirror-like reflection, which only contributes if $\mathbf{V} = \mathbf{M}$.

For this simplified BRDF, the integral in Equation (2) evaluates analytically to [17]

$$\mathbf{R} = - \left[(C_a + C_d) \mathbf{L} + \frac{2}{3} C_d \mathbf{N} + 2 \cos \theta_i C_s \mathbf{N} \right]. \quad (4)$$

If the target is in thermodynamic equilibrium, all absorbed radiation is reradiated instantaneously by Kirchhoff's law.

If this reradiation is Lambertian, the reaction vector becomes [17]

$$\mathbf{R} = - \left[(C_a + C_d) \left(\mathbf{L} + \frac{2}{3} \mathbf{N} \right) + 2 \cos \theta_i C_s \mathbf{N} \right]. \quad (5)$$

The specular contribution is strictly along the surface normal direction since its tangential components cancel. The Lambertian diffuse contribution (both reflected and reradiated) has a component along the incoming direction but also, weighted by a factor 2/3 (see [18] for a derivation of this factor), a component along the surface normal. The reaction vector is thus always in the plane spanned by \mathbf{L} and \mathbf{N} .

2.3 Radiation sources

Radiation sources emit or reflect radiation, which exerts RP onto the target. As explained in Section 2.1, the incident radiation at a target due to a source can be thought of as light rays, which are described by their directional irradiance at the target. How the directional irradiance is calculated depends on the type of source.

Isotropic point sources The simplest source model is a point source that isotropically radiates in all directions. This model is appropriate for far-away sources such as the Sun at 1 au distance. Due to the distance, all rays are effectively parallel and can be merged into a single ray parallel to the source-to-target vector $\mathbf{r}_{t/s}$. For an isotropic source, the total luminosity L (units of W) is uniformly distributed over a sphere, leading to an inverse square law. Therefore, the irradiance at the target is

$$E = \frac{L}{4\pi \|\mathbf{r}_{t/s}\|^2}. \quad (6)$$

Alternatively, a reference irradiance E_{ref} observed at a distance r_{ref} can be scaled:

$$E = E_{\text{ref}} \frac{r_{\text{ref}}}{\|\mathbf{r}_{t/s}\|^2}. \quad (7)$$

The solar luminosity is 3.828×10^{26} W [19], which corresponds to an irradiance of 1361 W/m^2 at 1 au. Note that these values are averages, which vary with the 11-year solar cycle by about 0.1% and more on shorter timescales due to sunspot darkening and facular brightening [20]. Observational time series exist to account for these variations [21].

Paneled sources: Discretization Radiation due to planets and moons requires more involved source models. Planetary emissions comprise reflected solar radiation and thermal infrared radiation [22]. The fraction of reflected sunlight is called albedo² a ; the corresponding type is therefore also called albedo radiation. Thermal radiation is due to absorbed solar energy that is re-emitted in a delayed fashion. Observation time series of albedo and thermal fluxes exist for Earth [21], but physical modeling is required for the Moon.

Since planetary radiation is not isotropic and the spacecraft is typically much closer to the body than to the Sun, the source extent has to be considered. In contrast to the previously described point source, we therefore model Earth

and Moon as extended sources. These are discretized into sub-sources, from which rays emanate that are, in general, not parallel. The sub-sources can be thought of as panels with an area, orientation, position, and radiosity model. The panel extent is represented by the area but any other panel properties are only evaluated at its center. A panel only radiates from the positive normal side, not from the backside.

Different algorithms exist to divide the planet ellipsoid into panels. Some authors use a longitude-latitude grid (e.g., [24, 25]), particularly with observed fluxes) or generate static, uniformly spaced panels over the whole sphere (e.g., [16]). However, both approaches are inefficient for low-altitude spacecraft, which require a large number of panels, most of which are never visible. Therefore, the de-facto standard is the dynamic³ paneling method introduced by Knocke *et al.* [22].

In Knocke's method, only the visible area of the planet is paneled. This area is a spherical cap, centered at the subsatellite point and divided into concentric rings that are, again, divided into equal-area segments. A central panel is located at the subsatellite point. All panels contribute to the irradiance received by the target. However, the effective area of each panel is projected by its viewing angle θ_r (see Figure 1) and the irradiance is attenuated by an inverse square law. In Knocke's method, the rings are spaced such that each panel has the same projected, attenuated area. The projected, attenuated area of a panel is defined as [22]

$$\frac{dA \cos \theta_r}{\|\mathbf{r}_{t/s}\|^2}, \quad (8)$$

where dA is the geometric panel area and $r_{t/s}$ is the source-to-target vector (in this case, the panel-to-target vector). More rings and more panels per ring improve the fidelity of the calculated irradiance, barring the resolution limit of the radiosity model (e.g., the albedo distribution). While arbitrary numbers of panels per ring are possible, Knocke suggests multiples of 6 (i.e., six panels in the first ring, twelve panels in the second ring, ...). The algorithm is elaborated in [26].

Two examples at different spacecraft altitudes and with different ring numbers are shown in Figure 2. At higher altitudes, a larger area is visible (approaching a hemisphere) and panels are somewhat more uniform in area. At lower altitudes, the panels are more tightly spaced towards the subsatellite point. In both cases, panel areas increase towards the edge of the visible cap. This pattern is result of the equal projected, attenuated areas.

Paneled sources: Radiosity models The emitted and reflected fluxes of a panel are described by a radiosity model. The irradiance at the target position can then be derived from the panel radiosity. Both radiosity and irradiance commonly have units of W/m^2 . Each panel can have one or more

²Two types of albedo exist: spherical/Bond albedo is the fraction of sunlight reflected in all directions, while geometrical albedo is the fraction of sunlight reflected with respect to an ideal diffuse surface for normal incidence and viewing directions [23]. For our purpose, spherical albedo is appropriate and synonymous with albedo in this paper.

³Dynamic refers to the fact that panels move with the spacecraft, as opposed to static paneling, for which panels are invariant with spacecraft position or time.

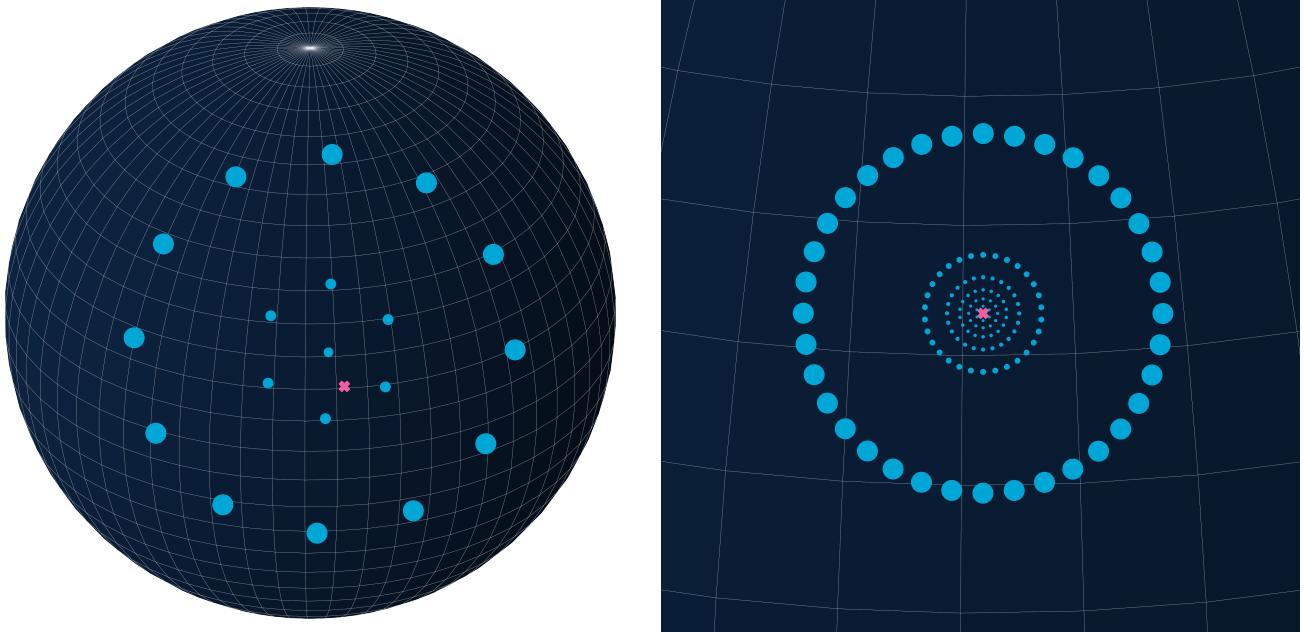


Figure 2. Panels generated with Knocke's algorithm for the Moon, which has a mean radius of 1737 km. The spacecraft (✖) sees a spherical cap (—), which contains rings of panels and is larger at higher altitudes h . Panel centers (●) are scaled proportional to the panel area. The panels have equal projected, attenuated areas and are therefore concentrated around the subsatellite point. The scenario in **b** corresponds to LRO's orbit and the paneling used in this paper.

radiosity model, usually one for albedo radiation and one for thermal radiation. We present three such models.

The albedo radiosity model accounts for diffuse Lambertian reflection of solar radiation. It implements the specular-diffuse BRDF from Equation (3) with $C_s = 0$ and the albedo value $C_d = a$ at the panel center. The albedo radiosity of a panel is [22]

$$J_{\text{albedo}} = a (\cos \theta_i)_+ E_s, \quad (9)$$

where E_s is the incoming solar irradiance at the panel (e.g., as found from Equation (6)) and the solar incidence angle θ_i is defined in Figure 1. The operator $(\cdot)_+$ restricts the input to positive values or zero otherwise. This ensures that no radiation is reflected from the backside.

The delayed thermal radiosity model assumes that absorbed radiation is emitted independently of incident solar radiation and the radiosity is thus not a function of θ_i . The only spatial variations arise from emissivity differences. The emissivity e of a surface is the ratio of the actual radiosity to the ideal black body radiosity. The delay arises from the planet's large thermal inertia. The delayed thermal radiosity of a panel is [22]

$$J_{\text{thermal}} = e \frac{E_s}{4}, \quad (10)$$

where e is the emissivity of the panel, evaluated at its center. The factor $1/4$ is the ratio of absorbing area (a circle) to emitting area (a sphere). The albedo and delayed thermal model were originally used by Knocke *et al.* for Earth emissions [22].

The angle-based thermal radiosity model is more appropriate than the delayed model if the surface experience significant diurnal cooling and heating. The surface

temperature is modeled as a function of the solar incidence angle θ_i and related to the radiosity through the Stefan-Boltzmann law. The surface temperature is interpolated between the minimum and maximum temperatures, T_{\min} and T_{\max} as

$$T = \max \left(T_{\max} (\cos \theta_i)_+^{1/4}, T_{\min} \right). \quad (11)$$

These temperatures typically correspond to the nighttime temperature and the temperature at the subsolar point. The angle-based thermal radiosity of a panel is then [27]

$$J_{\text{thermal}} = e \sigma T^4, \quad (12)$$

where T is the surface temperature from Equation (11) at the panel center and $\sigma = 5.670 \times 10^{-8} \text{ W}/(\text{m}^2 \text{ K}^4)$ is the Stefan-Boltzmann constant. On the dayside, the radiosity is proportional to $T_{\max}^4 \cos \theta_i$. The maximum radiosity of $e \sigma T_{\max}^4$ is usually larger than the near-constant $e E_s / 4$ from Equation (10), but quickly decreases as the panel moves away from the subsolar point (where $\theta_i = 0^\circ$). On the nightside, the thermal radiosity reduces to $e \sigma T_{\min}^4$.

The albedo and thermal radiosity models depend on the distribution of a and e over the planetary surface. The values may be assumed constant but generally vary with longitude, latitude, and time. Particularly for Earth, seasons and weather greatly affect reflectivity and emissivity [28]. Since the Moon lacks seasons, distributions that only vary spatially are appropriate.

To obtain the irradiance at the target due to the panel radiosity, we assume that the emission follows Lambert's cosine law and account for the projected, attenuated area of

the source panel. The irradiance therefore is

$$E = \left(\sum_{J_i \in \mathcal{J}} J_i \right) \frac{dA (\cos \theta_r)_+}{\pi \| \mathbf{r}_{t/s} \|^2}, \quad (13)$$

where \mathcal{J} is the set of radiosities from any of the previous radiosity models. Usually, a panel has the albedo model and one thermal model. Here, the source-to-target vector $\mathbf{r}_{t/s}$ uses the panel center position, not the source body center. The direction $\hat{\mathbf{r}}_{t/s}$ of the corresponding directional irradiance $\mathbf{E} = E \hat{\mathbf{r}}_{t/s}$ is therefore not the same for each panel and thus considers the extent of the source. The radiosities J_i in Equation (13) can be summed since their radiation emanates from the same point, the panel center. Contrarily, the directional irradiances \mathbf{E} can generally not be summed since the their individual directions need to be retained: the reflectance model of the target may be sensitive to the incoming direction of each ray. Therefore, a set of directional radiances \mathcal{E} is handed to the RP target model for acceleration calculations.

2.4 Radiation pressure targets

A RP target is a body that is accelerated by RP. The target model governs how the incident irradiances from point sources and extended sources accelerate the target body.

Cannonball target In its simplest form, a target can be modeled as isotropic sphere, also referred to as cannonball. This sphere is characterized by a cross-sectional area A_c (independent of orientation), radiation pressure coefficient C_r (incorporating reflectivity and absorption coefficients), and mass m . Due to its isotropy, any lateral components cancel and the net acceleration is always along the source-to-target vector. The RP acceleration of a cannonball target is [1]

$$\mathbf{a} = C_r \frac{A_c}{m} \sum_{\mathbf{E}_j \in \mathcal{E}} \frac{\mathbf{E}_j}{c}, \quad (14)$$

where the sum is vectorial and $\sum \mathbf{E}_j / c$ is the total RP as described in Section 2.1. \mathcal{E} is the set of directional irradiances from any number of sources, both point (Equation (6)) and paneled (Equation (13)). The dependence on the area-to-mass ratio A_c/m is similar to drag accelerations. While the cannonball model cannot account for complex geometry, it is often used in orbit determination with C_r as estimated variable. Ray tracing of a detailed model can also be used to establish the evolution of A_c and C_r [29].

Paneled target In reality, the cross-section and optical properties of a spacecraft change with orientation and incident direction. This effect is particularly noticeable for solar panels, which are large and usually track the Sun. To account for the geometry and differences in materials, a spacecraft can be represented as a collection of n panels. Each panel is characterized by its area, surface normal, and reflectance distribution. The position would only be relevant for rotational but not for linear accelerations. In case of moving parts, the surface normal may change over time. The reflectance distribution can be given as generic BRDF, but is often a specular-diffuse BRDF. The RP acceleration of a

paneled target is [30]

$$\mathbf{a} = \frac{1}{m} \sum_{\mathbf{E}_j \in \mathcal{E}} \left(\frac{\| \mathbf{E}_j \|}{c} \sum_{k=1}^n A_k (\cos \theta_{i,k})_+ \mathbf{R}_k \right), \quad (15)$$

where the indices j and k denote the (sub-)source and target panel, respectively. A_k is the area of the k -th panel. $\theta_{i,k}$ is the incidence angle of \mathbf{E}_i onto the k -th panel. \mathbf{R}_k is the reaction vector as defined by Equations (2), (4) or (5), depending on the BRDF. The reaction vector is a function of the panel surface normal \mathbf{N} and the source-to-target direction $\mathbf{L} = \hat{\mathbf{E}}_j$. Therefore, the inner sum has to be evaluated for each directional irradiance \mathbf{E}_j of the outer sum. However, \mathbf{E}_j itself is only calculated once for all panels at the target center to avoid quadratic computational complexity. In general, the resulting acceleration is not along the source-to-target direction as for the cannonball.

Extensions for the paneled target model exist. The model described above does not account for self-shadowing, which occurs when one ray would intersect two panels. This effectively reduces the area of the shadowed panel, an effect that can be significant for complex spacecraft geometries [31]. Polygon intersections enable simple calculation of the effective area [31]. Ray tracing is more involved but can also account for multiple reflections between target panels [32].

Another extension is the radiation pressure due to thermal radiation of the spacecraft itself. Instantaneous reradiation as modeled by Equation (5) for the case of thermodynamic equilibrium is a simple version of this. In reality, panels heat up and cool down (particularly during eclipses) through radiation, conduction, and internal heat production. Advanced models therefore calculate the temperature of each panel. Such models range from a simple heat balance [16] to finite element models [25]. However, lack of knowledge of the thermal properties may restrict the applicability. For the sake of simplicity, neither self-shadowing nor thermal radiation pressure of the spacecraft is considered in this paper.

2.5 Occultation

All previous models assume that the line of sight between source and target is unobstructed. However, occultation is a common astronomical phenomenon: a low-altitude spacecraft may be in the planet's shadow for more than a third of its orbit, and partial or full lunar eclipses can occur multiple times per year. We present two occultation models.

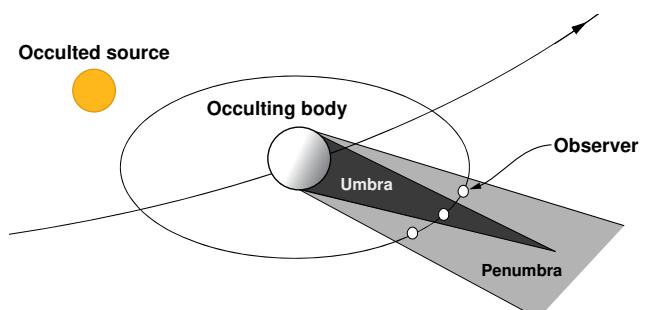


Figure 3. Conical occultation model for spherical sources and occulting bodies. The observer is partially illuminated in the penumbra but fully shadowed in the umbra. Adapted from [33].

Shadow function The shadow function ν describes the fraction of light received from a spherical source in the presence of an occulting spherical body. The geometry of the conical occultation model is shown in Figure 3. In the umbra, the source is fully occulted and the observer does not receive any radiation ($\nu = 0$), a state referred to as total eclipse. In the penumbra, the observer can see part of the source ($0 < \nu < 1$). Only outside the shadow region does the observer receive the full radiation ($\nu = 1$). In the case of a lunar eclipse, Earth occults the Sun and casts a shadow onto the Moon such that there is no lunar albedo radiation. On the nightside of a planet, the planet itself occults the Sun.

With the models described in Sections 2.3 and 2.4, the shadow function needs to be considered for radiation from a point source, both when directly incident on the target and when used as solar radiation for albedo radiosity. The extent of the source and occulting bodies needs to be known for shadow function calculations, even in the case of point sources. A derivation of the conical model for ν is presented by Montenbruck and Gill [1].

The conical model can only account for one occulting body. In case of multiple occulting bodies, shadows might overlap and the product of their shadow functions would underestimate the actual received fraction. Knowledge of the shadow intersection would be required to avoid this. Zhang *et al.* derived a model for two occulting bodies [34]. However, only single occultations are considered in this paper.

More involved shadow models exist that improve prediction of the penumbra passage. These models can consider planetary oblateness and atmospheric effects like absorption, scattering and refraction [35]. Other models can account for topography by combining a paneled Sun model with a topography map [9]. These modifications usually prolong the penumbra duration.

Point-to-point visibility For source panels represented by their center point, the shadow function becomes binary: either there is a line of sight between the panel center and the target or there is not. Such point-to-point visibility with a spherical occulting body is easily modeled geometrically. A derivation is given by Vallado and Wertz [33]. Multiple occultations are supported in this occultation model by the logical conjunction of the individual visibilities.

3 Radiation pressure modeling for LRO

3.1 Lunar albedo radiation

The Moon is a major source of radiation in LRO's orbit, with lunar irradiance magnitudes approaching half of the Sun's. Therefore, albedo and thermal radiation due to the Moon is modeled. While the lunar albedo is only 40 % of Earth's albedo [28], albedo radiation due to the Moon is still substantial, particularly over the subsolar point [6]. Lunar albedo varies significantly with geology: the highlands (mean $a = 0.16$, maximum $a = 0.25$) are much more reflective than the maria (mean $a = 0.07$, minimum $a = 0.05$) due to their respective regolith composition [37–39]. The mosaic of calibrated albedo imagery from Clementine in Figure 4a clearly shows the differences between highlands and maria. The mean of 0.12 agrees with other literature [37], and most of the lunar surface has an albedo below 0.20. Higher values

are only found at the poles, where the imagery represents topographic shading rather than actual albedo [40]. Note that the mosaic is for albedo of light at 750 nm wavelength, which is slightly longer than the average solar wavelength. Even though solar radiation has most energy within the 300 nm to 2400 nm band, the spectrum peaks at around 470 nm [41]. Lunar reflectivity increases with increasing wavelength [42].

Floberghagen *et al.*'s 15×15 spherical harmonics expansion called Delft Lunar Albedo Model 1 (DLAM-1) [6] is often used to represent this spatial albedo variability in lunar RP models. DLAM-1 was fitted from Clementine imagery and was designed to work with Knocke's albedo model for dynamic paneling (Equation (9)). Due to the nature of spherical harmonics, the model cannot resolve features smaller than 12° (360 km at the equator). The expansion is shown in Figure 4b, along with direct imagery from Clementine. DLAM-1 was also derived from 750 nm imagery, but we scale the original values by 1/1.3 to account for the reduced reflectivity at the average solar solar wavelength. This factor was proposed by Vasavada *et al.* [37]. Even with the correction, the mean albedo of the expansion of 0.15 is still 25 % above the commonly accepted mean of 0.12. This is possibly due to a different calibration of the imagery that DLAM-1 is based on compared to the mosaic from Figure 4a. In fact, Clementine is known to overestimate albedo due to bad calibration [42]. Apart from the difference in magnitude, the patterns agree reasonably well: maria and highlands are distinct and large bright features like the ray system around the Tycho and Giordano Bruno craters can be recognized (marked in Figure 4b).

Despite the shortcomings of DLAM-1, spherical harmonics are convenient: they are smooth and do not require interpolation like a gridded map. They can easily be truncated to trade detail for computational efficiency. Therefore, we used DLAM-1 in this paper but consider that the magnitude may be overestimated by 25 % during the analysis of results. We also compare results for the location-dependent DLAM-1 with those for a constant value, which should be more computationally efficient. As single representative albedo, we choose the mean of 0.15 instead of 0.12 to facilitate comparison. Note that the spatial variability described above suggests that a single albedo value cannot accurately represent lunar reflectivity.

Albedo radiation assumes ideal, diffuse Lambertian reflectance, which decreases with the cosine of the viewing angle. This assumption is especially appropriate for Earth, for which purely specular radiosity only amounts to 10 % of the purely diffuse radiosity [22]. However, this is not the case for the Moon: the opposition effect increases the reflectance at low phase angles (when the source is behind the observer, see Figure 3) much more than would be expected from a cosine law. In fact, the brightness increases more than 40 % between phase angles of 4° and 0° [43]. This is primarily caused by shadow hiding. To account for non-diffuse reflectance of the lunar surface, the Hapke BRDF was developed [44]. This BRDF is an empirical relation based on nine parameters that control, among other phenomena, the strength and directionality of the opposition effect. Near-global maps for these parameters have been fitted from LRO observations and could be used for a radiosity model [39]. For RP acceleration modeling, the opposition effect is only

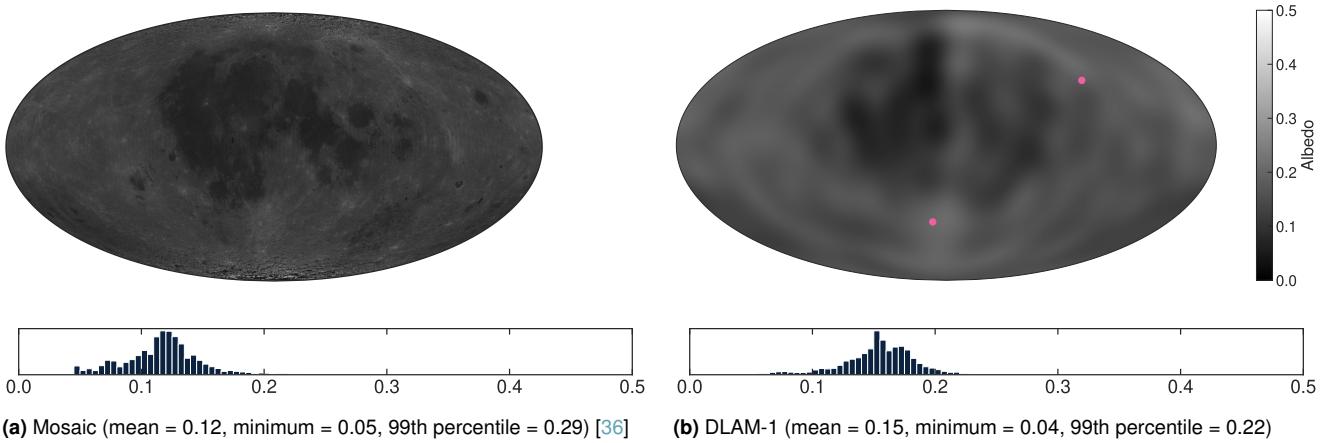


Figure 4. Lunar albedo distribution from Clementine. Both the mosaic and DLAM-1 are based on 750 nm reflectivity, but DLAM-1 has been corrected to the average solar wavelength. Large bright features like the Tycho and Giordano Bruno craters (●) can be registered. Note that the maximum of the albedo scale here is 0.5 instead of 1.0 to increase contrast; in reality, the Moon appears half as bright.

of concern when the target is above the subsolar point; the Sun has to be in the orbital plane for this. For LRO, this only occurs for a few days twice a year, and even then only for a small fraction of the orbit. Therefore, we neglect the opposition effect in this study.

3.2 Lunar thermal radiation

Lunar surface temperatures and the associated thermal radiation undergoes a significant diurnal cycle. Daytime and nighttime temperatures can differ by up to 290 K. The surface heats rapidly after sunrise, cools at about the same rate after local noon, then slower during the night [37]. There are small seasonal changes, with noon temperatures differing by 6 K between lunar aphelion and perihelion [23]. The large diurnal variability makes Knocke's delayed thermal model (Equation (10)), which gives a constant radiosity throughout the day, unsuitable for the Moon.

Diurnal variability is represented well by the angle-based thermal model (Equation (12)). We parametrize the model with the equatorial temperatures just before sunrise ($T_{\min} = 95$ K) and at local noon ($T_{\max} = 385$ K). The model transitions to the nighttime temperature when the incidence angle $\theta_i \geq 89.8^\circ$. The temperatures span a slightly larger range than Lemoine *et al.* ($T_{\min} = 100$ K, $T_{\max} = 375$ K), who initially proposed the angle-based model. However, they agree with those used by Park *et al.* [45]. Note that Park *et al.*'s model is identical up to a factor 1/4 in the radiosity, which is incorrect.

While the albedo varies with location (see Section 3.1), the emissivity and other thermophysical properties are remarkably uniform [38]. This means that a constant emissivity is a fair assumption. We used a value of $e = 0.95$, which is the broadband daytime emissivity, although it decreases to 0.90 during the night [46]. However, we assume the constant daytime emissivity at all times.

The thermal surface radiosity J_{thermal} from the angle-based model with the aforementioned parameters is shown in Figure 5. The radiosity decreases with the cosine of the incidence angle and approaches negligible emissions of 6 W/m² at nighttime. The maximum radiosity, which occurs below the subsolar point (i.e., at local noon), is

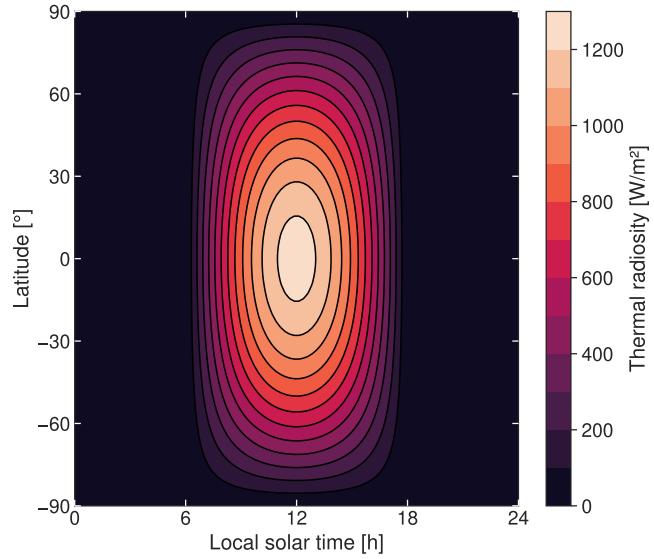


Figure 5. Map of lunar thermal emissions from the angle-based model (Equation (12)). The emissivity is 0.95 and surface temperatures range between 95 K and 385 K, depending on the subsolar angle.

1246 W/m². This peak value agrees with those used to design LRO's thermal control subsystem [2]. The only effect that is not captured is the slow cooling by about 25 K between sunset and sunrise [37], which introduces a slight asymmetry; constant pre-sunrise temperatures are used throughout the night. We also do not model seasonal variations of surface temperature.

3.3 Paneling of the Moon

The Moon needs to be discretized to evaluate the albedo and thermal radiation numerically. LRO's low altitude compared to the lunar radius prohibits any static paneling, which would have a large number of invisible panels. Therefore, we used Knocke's dynamic paneling method (Section 2.3).

Selecting the number of rings is a trade-off between fidelity and computational efficiency. To determine the lowest number of rings that can still represent lunar radiation with sufficient accuracy, we investigated the convergence behavior.

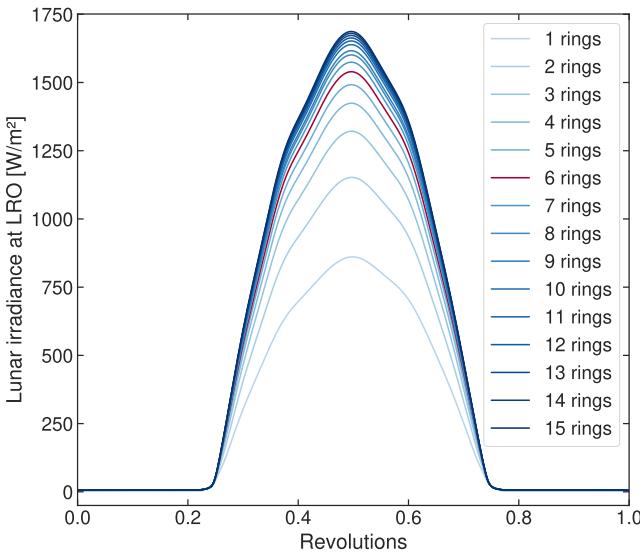


Figure 6. Convergence of lunar irradiance received by LRO for increasing number of rings. Each ring contains six more panels than the previous. Six rings (—) are sufficient for an error of less than 10 % with respect to the converged solution.

Figure 6 shows the albedo and thermal irradiance E received by LRO for an increasing number of rings. As suggested by Knocke *et al.*, each ring contains six more panels than the previous one. For 13 rings and more, the peak irradiance is within 1 % of 1687 W/m^2 . For six rings, the irradiance peaks at 1540 W/m^2 , which is within 10 % of the converged solution. The results are similar for constant and DLAM-1 albedo.

We choose six rings as sufficiently accurate, which contain 127 panels in total (cf. Figure 2b). This is one ring (or 36 panels) more than used by others. Floberghagen *et al.* suggest five rings for Lunar Prospector, which has twice the orbital altitude of LRO and thus needs fewer rings (Knocke *et al.* used only two rings for a much higher altitude relative to the planetary radius). Five rings are also used for LRO's precision orbit determination [10]. We choose one ring more to keep the error due to paneling below 10 %. More may be required in case of a higher-resolution albedo distribution.

3.4 LRO target

LRO comprises a cubical bus with a large solar array and a protruding high gain antenna (Figure 7). The solar array can be gimbaled partially about the Y and Z axes such that it can track the Sun when it is within LRO's orbit plane; if not, the solar array is at a fixed 45° angle with the -Y bus side. The antenna points towards Earth whenever it is visible [9]. This leads to large variations in cross-section over time, and different sides are presented to solar and lunar radiation.

LRO can be modeled as paneled target (Equation (15)) to account for this variability. The panels are summarized in Table 1. There are six panels for the bus, with surface normals along the positive and negative axes of the LRO bus frame (cf. Figure 7). The solar array and high gain antenna are modeled separately, again with frontside and backside panels. The solar array is almost as large as all other panels combined. Panels for the solar array and high gain antenna track the Sun and Earth or have a fixed orientation, as described above.

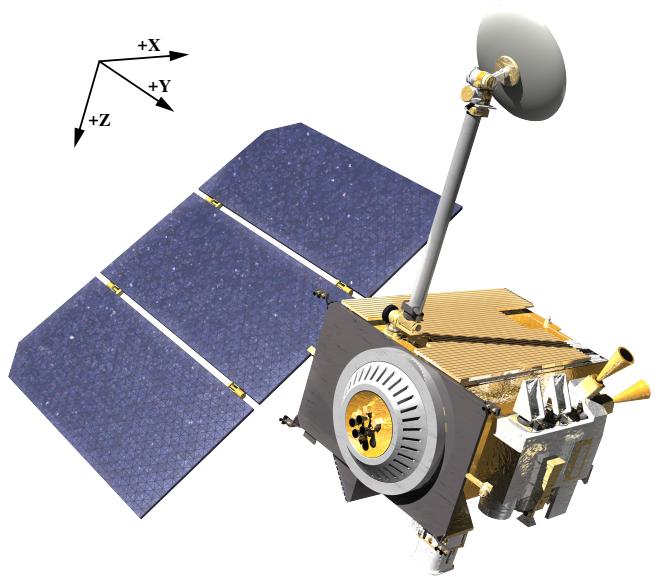


Figure 7. Rendering of LRO [47] with bus frame definition. The X axis is along the velocity vector, the +Y axis is away from the sun, and the +Z axis is in the nadir direction [2].

Definitive attitudes of actuated panels are also available but were not used.

We also model LRO as a cannonball (Equation (14)), a model which is often used for orbit determination. Finding a single equivalent cross-section area A_c and coefficient C_r that hold at all times is virtually impossible [33]. Different values for LRO exist in literature: Bauer *et al.* used $A_c = 10 \text{ m}^2$ and $C_r = 1.2$ [12], while Nicholson *et al.* used $A_c = 14 \text{ m}^2$ and $C_r = 1.0$ [10]. Their acceleration should differ by about 15 %. We choose the latter since it is used for operational orbit estimation of LRO.

To complete the target model, the mass is required to convert forces into accelerations. LRO performed monthly station keeping maneuvers during its science mission phase, which reduced the initial mass after science orbit insertion from 1272 kg to 1087 kg (Figure 8). With every maneuver, 6.3 kg of propellant are expelled [48]. This increases accelerations by 15 % over the course of 21 months. To facilitate comparison and obtain worst-case results, we

Table 1. Panels for LRO target model from Smith *et al.* [11]. The coefficients are for absorptivity and specular/diffuse reflectivity. The solar array is by far the largest surface, followed by the Z-facing panels.

Panel	C_a	C_s	C_d	$A [\text{m}^2]$	Tracking
+X	0.49	0.29	0.22	2.82	
-X	0.42	0.39	0.19	2.82	
+Y	0.45	0.32	0.23	3.69	
-Y	0.50	0.32	0.18	3.69	
+Z	0.50	0.32	0.18	5.14	
-Z	0.28	0.54	0.18	5.14	
+SA	0.90	0.05	0.05	11.00	+Sun or fixed
-SA	0.50	0.30	0.20	11.00	-Sun or fixed
+HGA	0.54	0.18	0.28	1.00	+Earth
-HGA	0.93	0.02	0.05	1.00	-Earth

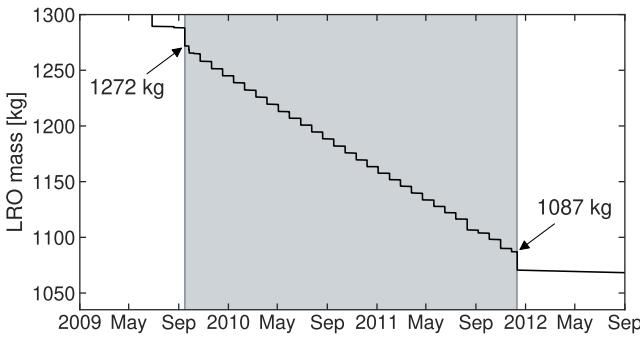


Figure 8. Mass evolution of LRO over the science mission phase (15 September 2009 – 11 December 2011, ■).

used the end-of-mission mass of 1087 kg in this paper, independently of the actual arc.

3.5 LRO orbit geometry

LRO's polar science mission orbit has a low altitude of 50 ± 15 km. The difference between periselene and aposelune is mostly due to a slight eccentricity, but also an equatorial radius that is 2 km larger than the polar radius. The altitude variation affects the lunar irradiance received by LRO. The spacecraft has a period of 113 min.

Visibility of the Sun is determined by the beta angle β , which is the angle between the orbital plane and the Moon-to-Sun vector. It is zero when the Sun is within the plane and $\pm 90^\circ$ when the orbit normal points towards or away from the Sun. Since LRO's orbit is polar, it undergoes all β within a year. For $\beta = 0^\circ$, LRO is moving straight towards and away from the Sun above the poles, experiences the maximum eclipse duration, and passes over the subsolar point. For $\beta = 90^\circ$, LRO is in full view of the Sun throughout the orbit and does not pass over hot or well-illuminated lunar regions. Far large β , LRO's solar array has a fixed orientation as mentioned in Section 3.4. Since the solar array then covers most of the -Y side, self-shadowing of up to 40% of the total cross-section can occur for solar radiation [9]. Self-shadowing is no issue for small β . In this paper, we neglect self-shadowing, which was shown to only have a minor impact in most cases [5, 49].

Since the effect of RP varies greatly with β , we investigated two contrasting arcs, which are summarized in Table 2. In June, Moon is at aphelion and no eclipses occur since LRO is continuously illuminated at $\beta \approx 90^\circ$. In September, the beta angle is just below 0° so that LRO experiences the maximum solar eclipse duration of 48 min. No lunar eclipses occur during any of the arcs. Both arcs have a length of 2.5

Table 2. Orbit geometry for selected arcs over 2.5 days. The June arc has permanent illumination, the September arc the maximum eclipse duration.

	28 June 2010	26 Sept. 2011
Start time (UTC)	15:00:00	18:00:00
β [$^\circ$]	88.8 to 88.9	-1.7 to -3.6
Sun distance [au]	1.019	1.000
Eclipse time [min]	0	48
Solar array	Fixed (45° to -Y)	Tracks Sun

days, which is used for orbit determination of LRO [8, 10]. Choosing the same length ensures our results are relevant for error estimation in force modeling for orbit determination.

3.6 Simulation setup

A range of simulations with varying models is necessary to determine the short-term orbital effects of RP modeling choices. However, a common setup, presented in Table 3, enables comparison. Some things should be clarified:

- Gravity from Jupiter and Venus is neglected since it is 7 orders of magnitude lower than Earth's.
- Albedo and thermal radiation due to Earth is neglected since it is 3 orders of magnitude lower than the Moon's.
- Occultation of solar radiation through the Moon follows a simple conical model. Time spent in sunlight can be overestimated by up to 480 s for large β if topography is ignored [9].
- No lunar eclipses occur over the two simulation arcs, therefore occultation by Earth can be ignored.
- Good accuracy can be achieved with integration step sizes as high as 15 s [9]. We use 5 s, which is used for operational orbit determination [10].
- While the Moon has a difference between polar and equatorial radius of 2.1 km due to flattening, a perfect sphere will be assumed here to simplify paneling and occultation.
- The Moon Mean Earth/Polar Axis frame is recommended for use with LRO [52].
- LRO SPICE kernels containing ephemerides (SPK) and orientation (CK) are available online⁴.

4 Results

We analyzed short-term effects of RP on accelerations and the position at the end of the arc. While the position is ultimately relevant for precise orbit determination, examining the accelerations in different scenarios and along the orbit can explain the cause of position changes. Additionally, the accelerations highlight differences between models of varying complexity.

To compare accelerations over one or multiple orbits, we used the root mean square error (RMSE), which is defined as

$$\text{RMSE}(x, y) = \sqrt{\frac{1}{n} \sum_{i=1}^n (x_i - y_i)^2}. \quad (16)$$

The RMSE describes the difference between two scalar timeseries x_i, y_i and gives more weight to large deviations. These scalar timeseries can be the magnitude of accelerations or individual components. The relative root mean square error (rRMSE) is defined as

$$\text{rRMSE}(x, y) = \sqrt{\frac{\sum_{i=1}^n (x_i - y_i)^2}{\sum_{i=1}^n y_i^2}} \quad (17)$$

and is useful to compare differences across orders of magnitude.

⁴https://naif.jpl.nasa.gov/pub/naif/pds/data/lro-l-spice-6-v1.0/lrosp_1000/

Table 3. Common setup for all simulations. Gravity and radiation pressure are the only force models.

Planetary bodies	
Planetary ephemerides	DE 421 [50]
Moon ellipsoid	Sphere of radius 1737.4 km [51] (no flattening)
Moon reference frame	Mean Earth/Polar Axis [52]
Force models	
Moon gravity	GRGM1200L [53] (truncated to 100×100)
Earth + Sun gravity	Central
Solar radiation	Isotropic point source ($L = 3.828 \times 10^{26}$ W [19]) Occulted by Moon
Lunar radiation	Paneled source with 6 rings of 6, 12, 18, 24, 30, 36 panels Albedo: Constant ($a = 0.150$) or DLAM-1
RP target	Thermal: Angle-based ($e = 0.95$, $T_{\min} = 95$ K, $T_{\max} = 385$ K) Cannonball ($A_c = 14 \text{ m}^2$, $C_r = 1.0$) or Paneled (see Table 1) Paneled model with or without instantaneous reradiation Mass: 1087.0 kg (end of science mission) Orientation: LRO_SC_BUS frame from SPICE CK
Simulation settings	
Software	TU Delft Astrodynamics Toolbox (Tudat) 2.12.1.dev19
Propagation frame	ECLIPJ2000
Propagation method	Cowell
Integration method	Runge–Kutta–Fehlberg 7(8)
Step size	5 s (fixed)
Arc length	2.5 days (31.9 revolutions)
Initial state	Cartesian state from SPICE SPK (lrorg_*)

While the simulation evaluates accelerations in a global frame, the effect of accelerations on the orbit is best analyzed in a spacecraft-fixed coordinate system that is aligned with the orbital track. The RSW coordinate system is one such system, defined by the unit vectors [33]

$$\mathbf{R} = \frac{\mathbf{r}}{\|\mathbf{r}\|}, \quad \mathbf{W} = \frac{\mathbf{r} \times \mathbf{v}}{\|\mathbf{r} \times \mathbf{v}\|}, \quad \text{and} \quad \mathbf{S} = \mathbf{W} \times \mathbf{R}. \quad (18)$$

The radial component \mathbf{R} is aligned with the planetocentric position vector \mathbf{r} . The cross-track component \mathbf{W} is aligned with the angular momentum vector, or orbit plane normal, involving the linear velocity \mathbf{v} . The along-track component \mathbf{S} completes the right-handed coordinate system. Note that \mathbf{S} is generally not aligned with the velocity vector, only for circular orbits.

4.1 Instantaneous reradiation

First, we investigated the effect of instantaneous reradiation for the paneled target model. This increases the acceleration proportional to each panel's C_a , normal to the panel (cf. Equation (5)).

Figure 10 shows the absolute and relative differences between accelerations without and with instantaneous reradiation. In absolute terms, the radial and along-track components are impacted most for the September arc, while the along-track and cross-track components experience the largest increase for the June arc (for both arcs, up to about $1.9 \times 10^{-8} \text{ m/s}^2$ RMSE). The relative differences are more uniform (around 40 % rRMSE), but the along-track components of lunar and solar radiation in the June arc increase by 140 % and 570 % rRMSE, respectively. In most

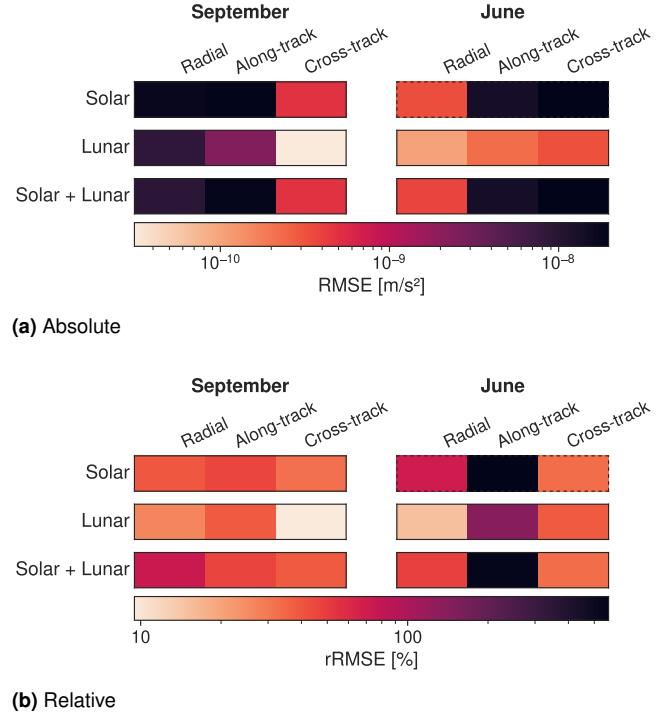


Figure 9. RMS differences of RP accelerations over one orbit with and without instantaneous reradiation. The dashed boxes correspond to Figure 10.

cases, only the magnitude of accelerations changes but not their pattern.

Figure 10 shows the solar radiation of the June arc, the only of our simulations for which the pattern changed significantly.

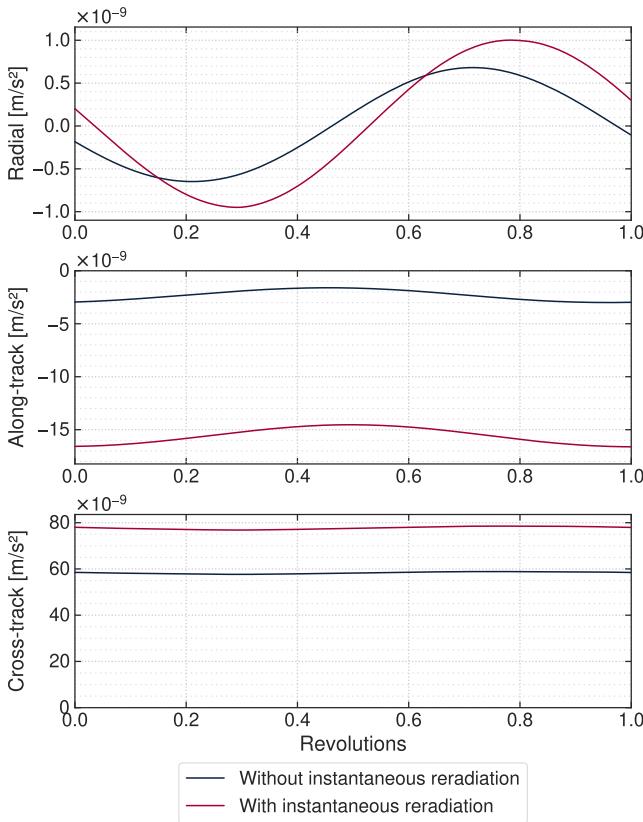


Figure 10. Accelerations due to solar radiation without and with instantaneous reradiation over one orbit for June arc. There is a phase shift in the radial component and the along-track component increased by 570 % RMSE. Lunar contributions and the September arc are not significantly affected in shape.

The phase of the radial acceleration is shifted by about 10 % of the orbital period, which is not the case for the other two components or the acceleration due to lunar radiation. This arc also had the largest relative change in along-track acceleration as described above (highlighted in Figure 9). This change is clearly visible as a constant offset of about $13 \times 10^{-9} \text{ m/s}^2$.

The large changes seen in some cases are mostly due to the +SA panel, which is highly absorptive ($C_a = 0.90$) and large ($A = 11.00 \text{ m}^2$). For the June arc, the solar array is angled at 45° with equal components in the cross-track and along-track directions. The Sun is on the same side as the solar array in the cross-track direction. Without instantaneous reradiation, no panel has a significant contribution to the along-track acceleration, so it is quite small at around $2 \times 10^{-9} \text{ m/s}^2$. With instantaneous reradiation, each panel, and especially the solar array, exerts an acceleration parallel to its normal, which leads to the along-track increase witnessed for the June arc.

We applied instantaneous reradiation for all of the following simulation since no reradiation due to spacecraft panels is physically unrealistic and the differences in magnitude are significant when instantaneous reradiation is added. More sophisticated thermal models involving conduction and internal heat production would likely give more accurate results.

4.2 Accelerations

The most direct effect of RP is visible in the accelerations. Therefore, we compare the RP accelerations

- for the September and June arcs,
- due to solar and lunar (albedo + thermal) radiation,
- for constant and DLAM-1 albedo distributions,
- for cannonball and paneled targets.

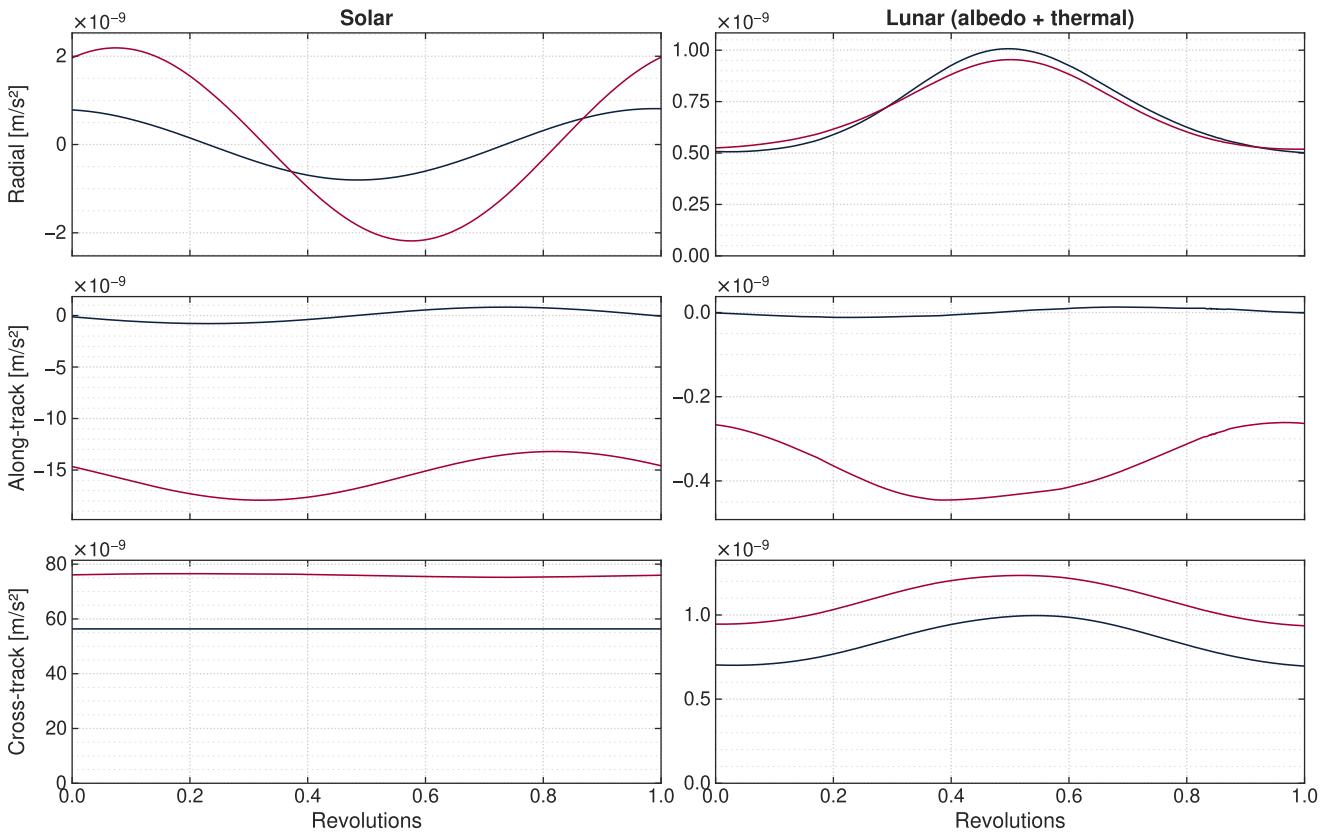
In total, we ran 46 simulations. All accelerations are given in 10^{-9} m/s^2 . Regarding cannonball and paneled target models, note that their comparative magnitudes are less important since the cannonball parameters have large uncertainties. Instead, we compared their behavior and how it relates to model assumptions (e.g., symmetry for the cannonball, tracking for the paneled target).

Solar and lunar radiation Both solar and lunar radiation are significant but their accelerations may amplify or cancel each other. To compare them, we used a constant albedo model in addition to the thermal model. The accelerations over one orbit are shown in Figure 11. Note that secular variations in orbit geometry can change the magnitude of acceleration components even within one 2.5 day arc but are not shown here.

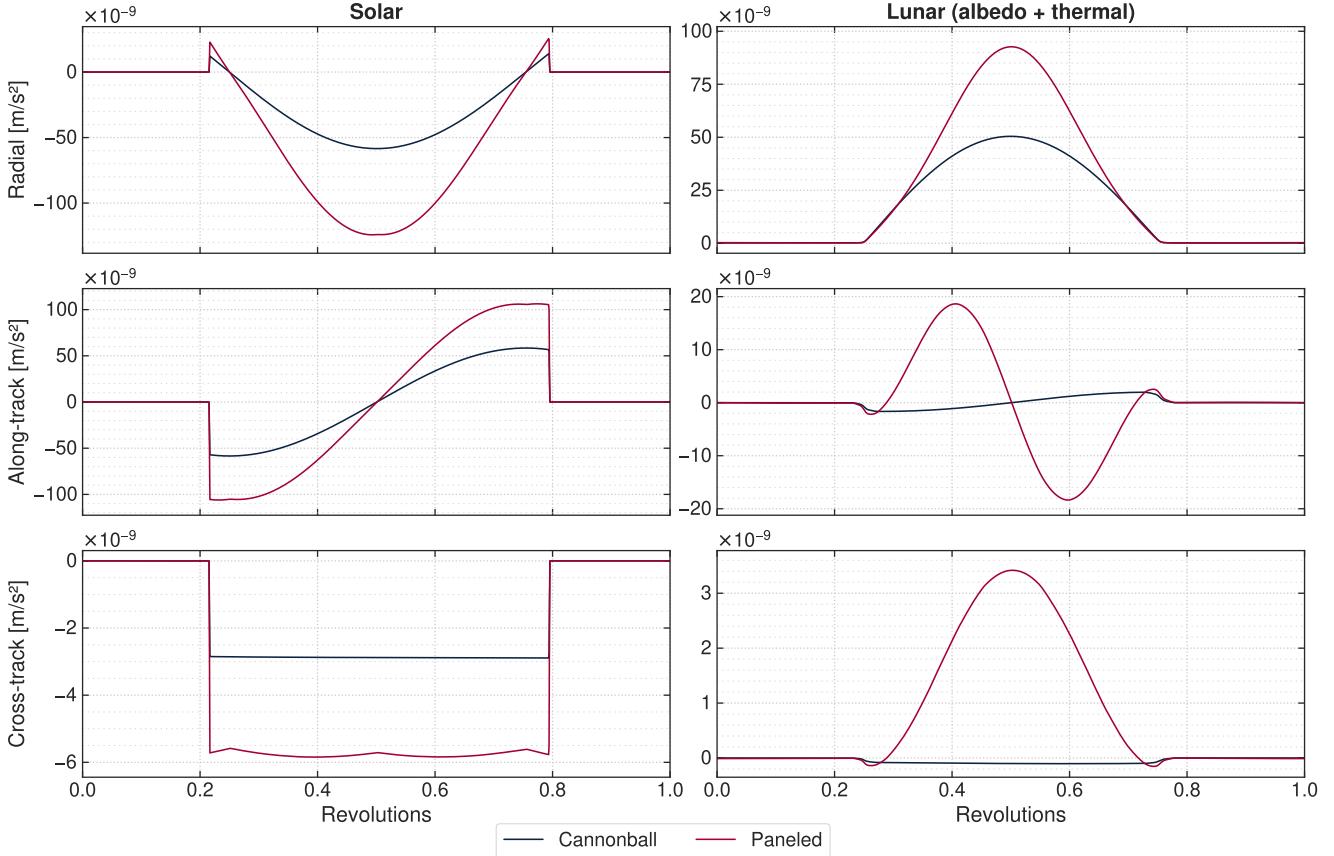
For the June arc (Figure 11a), the spacecraft is in permanent sunlight and orbit plane normal points towards the sun because $\beta \approx 90^\circ$. This leads to extremely large, constant cross-track solar accelerations. The paneled model also has along-track solar accelerations due to the solar array as explained in Section 4.1. Interestingly, the radial solar accelerations show the same phase shift between the cannonball and paneled target as observed without instantaneous reradiation (Section 4.1). This suggests that symmetry, or the lack thereof, is the cause of the phase shift. The magnitude of the total solar acceleration does not change much throughout the year since it is only dependent on the Moon–Sun distance, which is relatively constant at 1 au (see Table 2).

The lunar accelerations during the June arc are generally small (less than 2 % of solar) because LRO never passes over well-illuminated regions; half of the lunar source panels that are visible by LRO are on the nightside and therefore rarely contribute. The sinusoidal variations in lunar radiation pressure are mainly caused by the fact that β is not exactly 90° and LRO's angle to the subsolar point therefore varies by 2°. Secular variations in lunar accelerations (not shown) are due to the evolution of eccentricity over the 2.5 days caused by the non-uniform lunar gravity field [2]. The eccentricity ranges from 0.005 to 0.008, which leads to periselene altitudes between 37 km and 41 km. Changes in eccentricity lead to larger amplitudes but no mean shift. Periodic variations in altitude due to the eccentricity itself have a minimal effect since higher altitudes mean larger distances but also a larger visible area of the lunar surface, which roughly cancels.

For the September arc (Figure 11b), the Sun is occulted for 42 % of the orbit since $\beta \approx 0^\circ$. The effect of these occultations is clearly visible in solar and lunar radiation, both of which vanish on the nightside. The accelerations are mostly in the radial and along-track directions. This is most clearly explained with the solar accelerations. At 0.2, LRO



(a) June



(b) September

Figure 11. Accelerations due to solar and lunar radiation over one orbit for a paneled target. The cannonball and paneled target differ both in magnitude and pattern of accelerations. Note the different scales of each subplot.

crosses the terminator above the pole and is moving straight towards the Sun; the along-track component is maximal and negative since the Sun opposes the spacecraft's motion. Continuing the orbit, LRO passes above the subsolar point at 0.5, leading to a maximal and negative radial component while the along-track component has vanished. Further towards the other pole, LRO passes into the night at 0.8, where the along-track component accelerates the spacecraft into the direction of motion. During the whole time, the cross-track component is slightly negative because β is slightly negative but not zero. If β were slightly positive, the cross-track component would be similar but with the sign flipped. In addition to periodic changes, there is a secular change in solar accelerations (not shown) since the already slightly negative β continues to decrease: over the 2.5 day arc, the mean of the along-track and cross-track components increase twofold and threefold, respectively. This trend continues until the cross-track component dominates for high β , as seen for the June arc.

The lunar accelerations during the September arc are much larger than during the June arc because LRO passes right over the subsolar point, which reflects much sunlight and has high thermal emissions. Indeed, the lunar irradiance (up to 1831 W/m^2) is larger than the solar irradiance (up to 1362 W/m^2) above the subsolar point. Still, the lunar acceleration magnitude is 14% smaller than the solar acceleration magnitude since lunar panels are distributed azimuthally around the nadir and thus partially cancel while all solar rays are parallel. Another feature of lunar accelerations is the sign opposite to solar accelerations: when passing over the subsolar point, the radial components of solar and lunar accelerations roughly cancel. A similar effect can be seen in the along-track and cross-track components for a paneled target, although the lunar radiation needs some time to build up. The along-track component peaks at a subsolar angle of 33° , the cross-track component above the subsolar point. The cross-track component increases secularly over the 2.5 day arc, similarly to solar radiation.

Comparing the accelerations of cannonball and paneled targets for both arcs, it is clear that a single C_r cannot capture the complex and changing spacecraft geometry. While the solar accelerations for the September arc are just off by about a constant factor, this is not the case for the June arc or any of the lunar accelerations. In fact, the sign may even be different, particularly for lunar along-track and cross-track accelerations in September. This is likely caused by the solar array tracking the Sun. On smaller scales, the effect of target panels of different size and reflective properties becoming visible as LRO revolves around the Moon can be seen in the shape of the solar cross-track accelerations of the September arc.

All accelerations are inversely proportional to the spacecraft mass. While we chose the end-of-mission mass for all simulations, the begin-of-mission mass is 17% higher and all accelerations are thus 17% lower (see Section 3.4). This only changes magnitudes, not patterns.

Lunar albedo and thermal radiation In the previous subsection, lunar radiation was considered as the sum of albedo and thermal radiation. In this subsection, we look at the separate contributions and the differences between albedo

distributions. The accelerations on a paneled target are shown in Figure 12.

For both arcs and all components, thermal radiation is far larger than albedo radiation (up to sixfold). This is even though the albedo is likely overestimated by 25% as described in Section 3.1. In terms of behavior, the thermal radiation and constant albedo radiation are very similar: smooth and dependent on the subsolar angle. However, albedo radiation vanishes in the eclipse region of the September arc. The thermal irradiance at LRO on the nightside is 6 W/m^2 , which gives a small total acceleration of $1 \times 10^{-9} \text{ m/s}^2$.

The accelerations constant and DLAM-1 albedo distributions are of very similar magnitude (in non-zero regions, rRMSE of 19% for June and 24% for September), although DLAM-1 exhibits irregular variations. The largest difference of $5 \times 10^{-9} \text{ m/s}^2$ occurs in the radial component for the September arc. Interestingly, the maximum with DLAM-1 occurs before the subsolar point, where the thermal and constant albedo peak. This behavior can be explained by the ground track of this orbit. Figure 13 shows the DLAM-1 albedo irradiance over a map of the difference between DLAM-1 and the constant albedo. In this figure, the maximum also appears before the subsolar point and coincides with a region where DLAM-1 gives an albedo that is 0.067 higher or 55% more reflective than the constant albedo of $a = 0.150$. All irregular differences between the albedo distributions are explained by this map. However, because the thermal radiation is so much larger, the effect of these differences remains overall limited.

4.3 Change in final position

The goal of orbit determination is to estimate the state, and particularly the position, of a spacecraft. Therefore, the difference in position at the end of the arc between models is highly relevant. Since the true position is not known and this paper is rather concerned with relative differences, we use a simulation without solar and lunar RP as baseline reference. As mentioned in Section 1, the maximum allowable error is 50 m to 100 m in total position and below 1 m radially.

The difference in final positions with respect to the baseline simulation are shown in Table 4. The first number is the mean secular difference over the final orbit (32nd revolution), the second number gives the amplitude of periodic variations around that mean over the final orbit. Note that the periodic variation is quite large in some cases despite zero secular change.

The June arc shows a large along-track difference (more than 1 km) for the paneled target with solar radiation. This is due to the consistently large along-track acceleration of about $-15 \times 10^{-9} \text{ m/s}^2$, which, however, is lower and of opposite sign than the constant acceleration of $47 \times 10^{-9} \text{ m/s}^2$ expected to result in this position difference. For the cannonball, the along-track accelerations have a zero mean and thus the position difference is also zero. This, again, highlights how the cannonball cannot account for asymmetry; with symmetric accelerations, no secular changes occur. Interestingly, the large cross-track acceleration does not lead to a large cross-track difference in final position. Possibly, along-track accelerations have the greatest effect since they

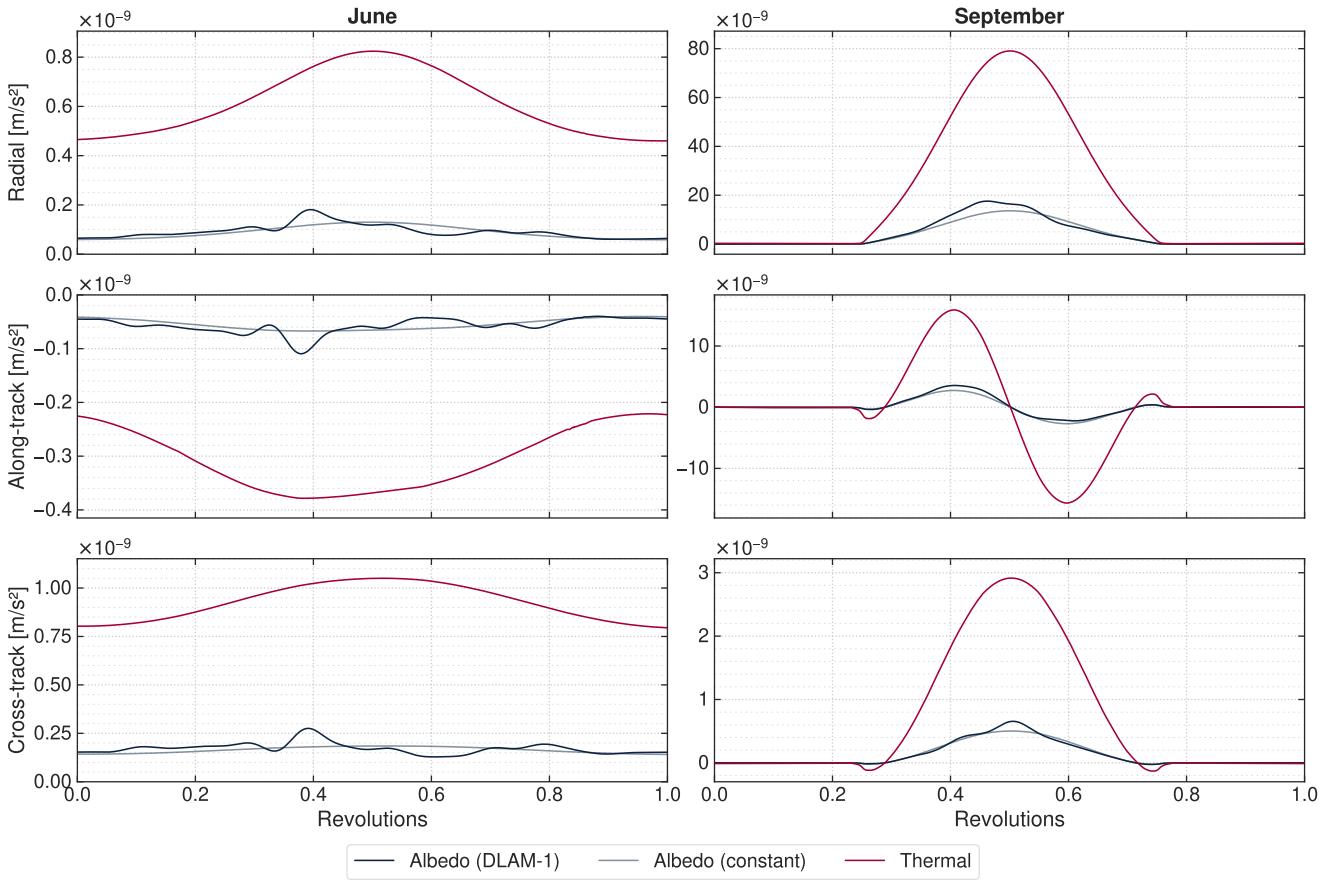


Figure 12. Accelerations due to lunar thermal and albedo radiation on a paneled target. Thermal radiation dominates at all times and the difference between constant and DLAM-1 albedo is small. Note the different scales of each subplot.

Table 4. Difference of final position in m with respect to no-RP baseline, given as mean over the final orbit plus/minus periodic variations around that mean in the final orbit. The largest changes are in the along-track position. A: solar only; B: lunar only (thermal + constant albedo); C: lunar only (thermal + DLAM-1 albedo); D: solar + lunar (thermal + DLAM-1 albedo).

Cannonball				Paneled			
Radial	Along-track	Cross-track	RMSE	Radial	Along-track	Cross-track	RMSE
A $+0.0 \pm 0.2$	-0.5 ± 0.5	$+0.1 \pm 0.1$	0.6	-7.5 ± 6.7	$+1066.1 \pm 39.3$	$+0.1 \pm 0.2$	1033.5
B $+0.0 \pm 0.0$	-0.3 ± 0.0	$+0.0 \pm 0.0$	0.3	-0.2 ± 0.2	$+24.4 \pm 0.9$	$+0.0 \pm 0.0$	23.6
C $+0.0 \pm 0.0$	-0.3 ± 0.0	$+0.0 \pm 0.0$	0.3	-0.2 ± 0.2	$+24.6 \pm 0.9$	$+0.0 \pm 0.0$	23.9
D $+0.0 \pm 0.2$	-0.8 ± 0.4	$+0.1 \pm 0.1$	0.8	-7.7 ± 6.9	$+1090.8 \pm 40.2$	$+0.1 \pm 0.2$	1057.4

(a) June

Cannonball				Paneled			
Radial	Along-track	Cross-track	RMSE	Radial	Along-track	Cross-track	RMSE
A $+0.2 \pm 12.4$	-36.4 ± 25.1	$+0.0 \pm 0.5$	40.3	$+0.3 \pm 23.7$	-61.8 ± 47.7	$+0.0 \pm 0.9$	70.3
B $+0.1 \pm 2.9$	-12.2 ± 5.9	$+0.0 \pm 0.1$	12.7	$+0.1 \pm 6.1$	-11.6 ± 12.4	$+0.0 \pm 0.2$	14.6
C $+0.1 \pm 2.9$	-12.6 ± 6.0	$+0.0 \pm 0.1$	13.1	$+0.1 \pm 6.2$	-21.4 ± 12.9	$+0.0 \pm 0.2$	22.8
D $+0.2 \pm 9.5$	-49.1 ± 19.8	$+0.0 \pm 0.4$	50.0	$+0.4 \pm 17.5$	-83.2 ± 36.3	$+0.0 \pm 0.6$	85.2

(b) September

are almost aligned with the large orbital velocity (1.7 km/s). Constant and DLAM-1 albedo do not differ significantly.

For the September arc, cannonball and paneled target give more similar results. Again, the largest secular difference is in the along-track position. However, in contrast to the June arc, LRO is shifted back in the track this time. Due to the large variations in radial and along-track accelerations over each orbit, there are large periodic variations in the

radial and along-track positions too. For the paneled target with solar and lunar radiation, these have amplitudes of 18 m and 36 m for the radial and along-track differences, respectively. Interestingly, these periodic variations have higher amplitudes when lunar radiation is not considered, which would otherwise cancel solar accelerations partially. There also is a 8 m RMSE difference for the paneled model between constant and DLAM-1 albedo.

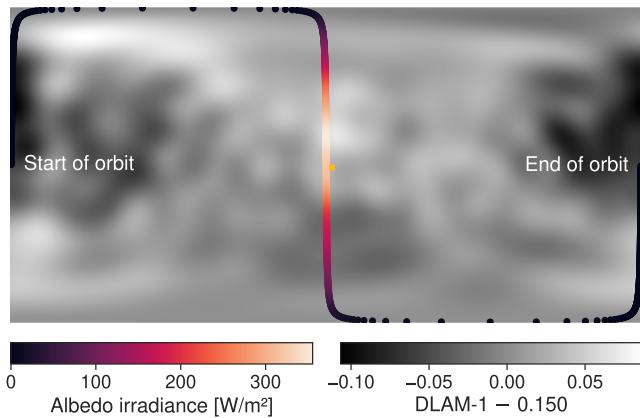


Figure 13. Ground track for September arc, colored by the irradiance due to DLAM-1 albedo. The map shows the difference between DLAM-1 and the constant albedo ($a = 0.150$). The maximum irradiance does not occur above the subsolar point (yellow star) but above the local albedo maximum 18° north of it.

These differences in position highlight how important RP models are for precise orbit determination. The best estimate of the true effect of RP is given by the cases D (solar + lunar radiation) for the paneled target. For both arcs, the maximum allowable total error would be exceeded by periodic variations if RP were neglected. The radial requirement of sub-meter accuracy would be similarly violated.

4.4 Performance

Choosing the appropriate model is not only a consideration of accuracy but also one of computational effort. The goal is to find the most efficient model that is sufficiently accurate. To determine how much less efficient complex models like paneled targets or DLAM-1 are, we measured the walltime duration of the integration over 2.5 days with a step size of 5 s. We ran each simulation 100 times in random order on a server with 2 Intel Xeon E5-2683 v3 (14 cores, two threads per core with hyperthreading) while no other loads were running. 27 simulations ran in parallel such that all but one core were used by a single thread. More parallelism would have triggered hyperthreading, which would have skewed measurements. The simulation program was compiled in the release configuration with GCC 11.4.0 at optimization level $-O2$. No other steps such as CPU pinning were taken. Note that software performance can be influenced by many, seemingly innocuous aspects [54]. Still, the results show the general tendency.

The walltime durations when including solar or lunar radiation are shown in Figure 14. Again, the baseline simulation without and RP models serves as reference. The median baseline time of 4.3 min includes computations of the lunar spherical harmonics gravity model, point gravities from Sun and Earth, and general integration and propagation equations. Solar radiation has a negligible complexity, even for a paneled target. Lunar radiation with constant albedo increases the duration by about 20 %, but slightly more for the paneled than the cannonball target. Taking the albedo from the spherical harmonics expansion of DLAM-1 increases the duration by up to 80 %, compared to the baseline. This is a significant impact which may not always be tolerable.

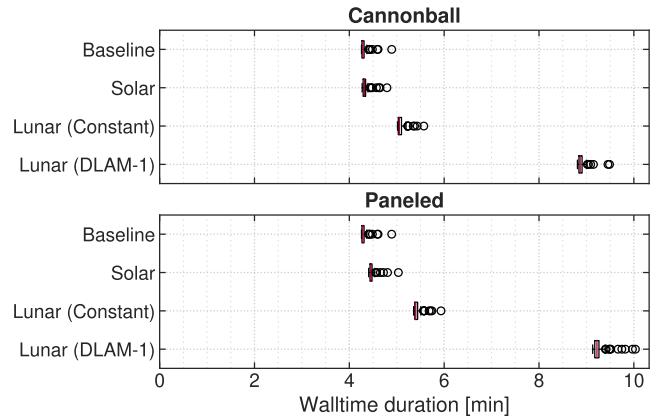


Figure 14. Walltime duration of simulations with different RP models. The statistics come from 100 runs for each model. Evaluation of DLAM-1's spherical harmonics expansion increases the duration by up to 80 %.

Other authors report increases of several hundred percent for DLAM-1 albedo [10].

The durations are relatively consistent so that the interquartile range is at most 5 s. Still, all distributions have a long tail towards higher durations: the difference between maximum and median is between 9 and 21 times higher than the difference between minimum and median. This behavior is typical for software.

5 Conclusion

We described a flexible framework of RP models of varying levels of complexity, then examined the differences in short-term orbital effects of RP on LRO between these models. Solar radiation follows a simple inverse square law, while the Moon has to be discretized into panels to evaluate lunar thermal and albedo variations. Thermal radiation dominates, and a constant albedo distribution is both sufficient and computationally cheaper than the spherical harmonics expansion DLAM-1. Large differences exist between the representations of LRO as a cannonball and a paneled target: due to the cannonball's symmetry, accelerations do not cancel over an orbit and are generally smaller than those of a paneled target, which can have the solar array track the sun. There are also large seasonal differences in the RP accelerations: for small β (e.g., around September), the accelerations are mainly radial and along-track, while they are predominantly cross-track for $\beta \approx \pm 90^\circ$ (e.g., around June). After 2.5 days, the position diverged from the no-RP baseline by up to 1100 m in June and 80 m in September. Periodic variations of up to 50 m are superimposed on the secular differences over the orbit. In September, the periodic variations are damped by lunar accelerations that oppose solar accelerations.

Our results showed that RP is essential for precise orbit determination of the LRO. Both the total and radial accuracy requirements would be violated otherwise. However, not all models are worth the computational effort. We recommend the following setup:

- Solar radiation should be included since it is cheap and important.

- Lunar thermal and constant albedo radiation should be included since it only increases walltime duration by 20 % and affects secular and periodic variations significantly. To reduce the performance impact, less rings could be used, although the lunar irradiance will then be underestimated.
- The spatially varying albedo distribution from DLAM-1 does increase the accuracy much even though it adds a performance penalty of 80 %. Therefore, it should not be included unless the utmost accuracy is desired (particularly in September). The spherical harmonics expansion could also be truncated to improve performance.
- The paneled target should be included since the cannonball underestimates accelerations and does not account for sun tracking of the solar array. The performance impact of the paneled target is negligible.
- The cannonball target should only be used if its coefficient is estimated since a constant coefficient cannot represent changes in geometry and orientation. Even then, consistent estimation of the coefficient is problematic at small β [7].

While we restricted our investigation to a small number of models, the short-term orbital effect of these more involved models should also be investigated:

- Moon topography can advance eclipse onset by up to 480 s for $\beta > 70^\circ$ [9]. The conical shadow model should be replaced by one that evaluates lines of sight based on topography.
- Self-shadowing, particularly for solar radiation, can reduce the effective cross-section by up to 40 % for large β [9].
- DLAM-1 was published in 1999 based on miscalibrated Clementine imagery, which overestimates the actual albedo. A new spherical harmonics model should be fitted from more recent imagery.
- Acceleration due to thermal reradiation by the spacecraft itself can be significant. Our model of instantaneous reradiation should be replaced by one that accounts for heating and conduction.
- The lunar opposition effect increases albedo radiation for low phase angles much more than Lambertian reflectance predicts, which becomes relevant for small β . The Hapke BRDF with spatially resolved parameter maps [39] should be used for albedo reflection.
- Post-sunset lunar thermal radiation is underestimated because gradual cooling to nighttime temperatures is not reflected in our thermal model. It should be replaced with a more physical model due to the large magnitude of thermal compared to albedo radiation.

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