

Assignment 2

$$1. \quad MR = \frac{m_f}{m_e} = \frac{m_p + m_c}{m_e} = \left(\frac{1}{1-r} \right)$$

where r is the ~~ratio~~ ^{fraction} we're supposed to find out.

Ideal rocket eqⁿ:

$$\Delta V = I_{sp} \cdot g_0 \cdot \ln(MR)$$

$$\therefore \Delta V = I_{sp} g_0 \ln \left(\frac{1}{1-r} \right)$$

$$\therefore \ln \left(\frac{1}{1-r} \right) = \frac{\Delta V}{I_{sp} g_0}$$

$$= \frac{7600}{400 \times 9.8}$$

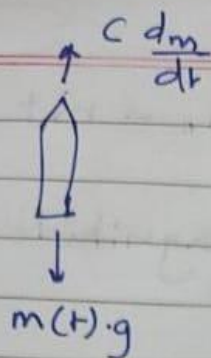
$$= 1.94$$

$$\therefore \frac{1}{1-r} = 6.95$$

$$\therefore r = 1 - \frac{1}{6.95}$$

$$= \boxed{0.856}$$

2.



The net force acting on the rocket in $+z$ dirⁿ, would be

$$\left(c \frac{dm}{dt} - m g \right)$$

The acc. would be

$$\frac{c}{m} \frac{dm}{dt} - g, \quad \text{thus}$$

$$\begin{aligned} v(t) &= \int \left(\frac{c}{m} \frac{dm}{dt} - g \right) dt \\ &= c \ln \left(\frac{m(t)}{m_0} \right) - gt + k \end{aligned}$$

Putting $t = 0$, we get $v = 0$, thus $k = 0$.

$$\begin{aligned} \therefore v(t) &= c \ln \left(\frac{m(t)}{m_0} \right) - gt \\ &= \boxed{c (\ln \mu) - gt} \end{aligned}$$

$$\begin{aligned} z(t) &= \int v(t) dt \\ &= \int (c \ln(m(t)) - c \ln m_0 - gt) dt \end{aligned}$$

We assume $\frac{dm}{dt}$ to be constant, which is the case in solid (and liquid) propellant rockets.

Thus $m(t) = m_0 - \dot{m}t$

(Here \dot{m} is the magnitude of the flow rate)

Note that

$$\int \ln(a-bx) dx = -\frac{a}{b} \ln(a-bx) + x \ln(a-bx) - x + d.$$

Thus

$$z(t) = c \left(-\frac{m_0}{\dot{m}} \ln(m_0 - \dot{m}t) + t \ln(m_0 - \dot{m}t) - t \right) - c(\ln m_0)t - \frac{gt^2}{2} + \frac{cm_0 \ln m_0}{\dot{m}}$$

~~$\frac{c}{\dot{m}}$~~ Now note that

$$m_0 - \dot{m}t = m(t) = \mu m_0$$

and $\dot{m} = \frac{m(t) - m_0}{t} = \frac{m_0 - \mu m_0}{t}$

$$= \frac{m_0 - \mu m_0}{t}$$

$$= \frac{m_0}{t} (1 - \mu).$$

$$\begin{aligned} \text{Thus } \frac{m_0}{m} &= \frac{m_0}{\frac{m_0 (1 - \mu)}{t}} \\ &= \frac{t}{1 - \mu}. \end{aligned}$$

Putting these expressions in our eqⁿ for $z(t)$,

$$z(t) = \left[c \left(t \ln(\mu m_0) - t - \frac{t}{1 - \mu} \ln(\mu m_0) \right) - c t \ln m_0 - \frac{g t^2}{2} + \frac{c t \ln m_0}{1 - \mu} \right]$$

We haven't been given the ratio of masses of propellant and dry mass, thus the total 'powered-flight' time would have an upper limit, given by

$$t_b = \frac{m_0}{\dot{m}} = \frac{m_0 c}{F_{\text{thrust}}} = \boxed{\frac{c}{g n}}$$

$$\text{Note that } I_{sp} = \frac{F_{\text{thrust}}}{\dot{m} g}$$

$$\text{Thus } n = \frac{F_{\text{thrust}}}{m_0 g} = \frac{F_{\text{thrust}}}{\dot{m} g t_b} = \frac{I_{sp}}{t_b}$$

i.e. the ratio of specific impulse and burning time.

