

# **CACTUS User's Manual**

## **Version 1.1**

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# 1 Introduction

This document serves as a user's manual for CACTUS (Code for Axial and Crossflow TURbine Simulation). A brief overview of the code is provided below, however, more detailed information on the methods used can be found in [1].

## 2 CACTUS Overview

CACTUS is a turbine performance simulation code based on the blade element method and using a free vortex line description of the turbine wake flow. The code was originally based on VDART3, a free vortex wake simulation of the Darrieus wind turbine, developed by Strickland [2]. The codebase has been largely upgraded to the Fortran 9x standard, and a number of modifications have been made to the original VDART3 methods including updates to the blade loads models, and new models to handle generic device geometry and marine turbine specific physics.

### 2.1 Blade Loads and Wake Models

CACTUS simulates a turbine device consisting of an arbitrary configuration of blade element sections. Each section can be assigned arbitrary load coefficient vs. angle of attack characteristics, which typically correspond to two-dimensional lift and drag coefficient data for a particular foil section. Since data from two-dimensional wind tunnel tests or foil performance calculations are used to represent element loads, it is generally assumed that these elements are in locally two-dimensional flow.

A rotor blade consisting of an arbitrary planform shape and foil sections can be modeled by the synthesis of a number of blade elements. The blade loads and wake of the turbine rotor are evolved in time over a certain number of rotor revolutions, until the revolution-averaged rotor power is converged. The code output includes the blade aerodynamic forces, wake vortex trajectories, and performance metrics such as torque and power.

CACTUS uses a potential flow model comprised of free vortex line elements to represent the turbine wake flow field. The vortex line structure attached to a single blade element is shown in Figure 1. At each point in time, the bound vorticity ( $\Gamma_B$ ) on each blade element is related to the element lift coefficient through the Kutta-Joukowski theorem, and the spanwise ( $\Gamma_S$ ) and trailing vorticity ( $\Gamma_T$ ) are recovered through the application of the Helmholtz theorem of conservation of circulation along a vortex line [3].

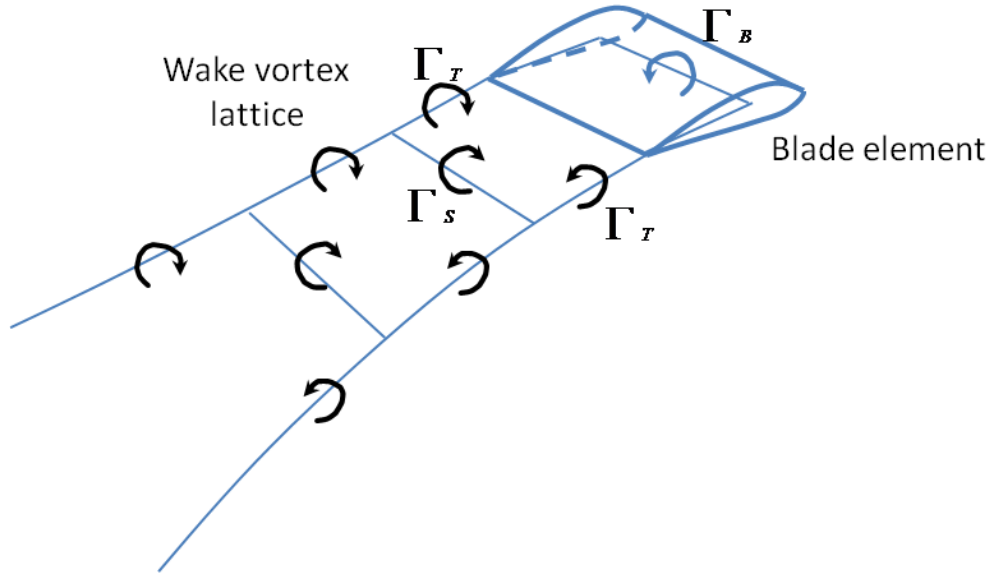


Figure 1. Blade element with associated vortex lattice system.

### 2.1.1 Dynamic Blade Loads

The operational cycle of some turbines, most notably cross-flow turbines or axial flow turbines in yawed flow conditions, cause the turbine blades to operate in dynamically variable flow conditions. The effects of blade rotation with respect to the surrounding fluid and the effects of dynamically variable flow angle of attack are captured with additional models.

The effects of blade section pitch rate (rotation around an axis normal to the section plane) are captured by analogy to an analytical solution for a pitching flat plate. Improvements have been made to the original methodology used in VDART3, and modifications have been made to handle non-zero section pitching moment due to cambered foil sections.

Under certain operational conditions, the turbine blades may operate at angles of attack beyond their steady-state stall limits for significant lengths of time. The transient behavior of the blade section loads during this “dynamic stalling” process must be modeled as it is not captured by the steady load coefficient data input for each foil section. The primary effect of dynamic stalling is a delay in the appearance of stalled flow effects on blade loads to higher angles of attack than would be expected in steady flow.

Two models for dynamic stall effects on blade section loads are included in CACTUS. The modified Boeing-Vertol method of Gormont is the default. This algebraic method approximates dynamic stall effects with a “lagged” angle of attack, where the magnitude of the lag is empirically correlated to the angle of attack rate. The Leishman-Beddoes model incorporates more physical models and attempts to model the temporal evolution of dynamic stall flow phenomena and associated effects on blade loads. This model may provide more accurate results than the algebraic Boeing-Vertol method, but requires many more simulation time steps to be taken per turbine revolution to achieve converged results.

## 2.2 Wall Boundaries

CACTUS can simulate the effects of proximity to a ground plane or free (water) surface on turbine performance. The boundary conditions, either zero normal flow for a ground plane or constant surface pressure for a free surface, are applied using rectangular source panel elements. The free surface boundary condition is currently implemented as a quasi-static boundary, allowing it to respond only to the average flow created by the turbine and wake over a full revolution.

The user is allowed to specify the time step interval between updates to the wall panel system. For the free surface model, the wall update interval specifies the number of time steps between updates to the revolution averaged quantities. It is often not necessary to update these quantities on every time step. If it's possible to reduce the frequency of wall panel system updates and still obtain convergence of the simulation output of interest, this can reduce simulation run time considerably.

## 3 Normalization Parameters

Some definitions of parameters used in normalize CACTUS input and output parameters are given below.

$\rho$	Fluid density
$U_{\infty}$	Freestream fluid flow speed
$A_T$	Turbine reference area. Typically this reference area is chosen to be the projected frontal area of the volume swept by the rotor.
$R$	Turbine reference radius
$\omega$	Turbine rotation rate
$U_{Loc}$	Fluid flow speed local to an element.
$U_{Tip}$	Turbine “tip” speed. Defined as $U_{Tip} = \omega R$ .
$c$	Element chord length
$A_E$	Element planform area

In general, most parameters have been normalized at the machine scale. Unless otherwise noted in the input and output descriptions below, the force, torque, and power coefficients are normalized as

$$\begin{aligned}
 C_F &= F / \frac{1}{2} \rho U_{\infty}^2 A_T \\
 C_T &= T / \frac{1}{2} \rho U_{\infty}^2 A_T R \\
 C_P &= P / \frac{1}{2} \rho U_{\infty}^3 A_T
 \end{aligned} \tag{1}$$

## 4 Input Description

This section describes the namelist, geometry, and foil data input files required by CACTUS. CACTUS is run from the command line with the path to the namelist input file passed as the only argument. The geometry input file and the foil data table files are referenced in the namelist input file.

### 4.1 Geometry Input

This section describes the CACTUS turbine geometry input file format, and the tools provided for generating an input file for a generic turbine configuration. CACTUS considers the turbine rotor to be comprised of lifting-line blades and optional support struts. Both blades and struts are decomposed into elements. The primary difference between the two is that the strut elements do not shed a free vortex wake and use simplified empirical models for element loads.

When generating geometry, it should be noted that the nominal freestream flow direction in CACTUS is the +x direction. If a ground plane or free surface calculation is being performed, these surfaces are oriented with their normal vectors in the +y direction.

The CACTUS geometry input file is comprised of a header section and individual sections for each of the blades and struts. Each blade and strut section contains the details of the element decomposition at the initial time in the turbine simulation. In general, the blade and strut geometry is specified by the user at the element end points. The element specific geometry maintained in the geometry input file can be calculated consistent with the element end geometry using the geometry creation tools described in Section 4.1.3.

The parameters and file format are described below. Note that each parameter occupies one line in the geometry file and array valued parameters are input as component values separated by spaces.

#### 4.1.1 File Parameters

##### 4.1.1.1 Header

$B$	Number of blades.
	Number of struts.
	Turbine rotation axis normal vector (x y z values).
	Turbine rotation origin point (x y z values).
$A$	Turbine reference area (for force/torque/power normalization) divided by reference radius squared.
	Turbine reference radius (reference length dimension) for scaling dimensional output values (ft). Corresponds to the reference radius used to normalize all geometry inputs below.
	String indicating the turbine geometry generation function used to create this file. This line is for reference only, and is not used internally in CACTUS.

#### 4.1.1.2 Blade

The blade geometry is specified by the user at the element end points in terms of the locations of the blade quarter-chord line, the local blade tangent vector components, and the local chord-to-radius ratio at the initial time of the turbine simulation. The element specific geometry should be calculated consistent with the element end geometry using the geometry creation tools described in Section 4.1.3. The parameters directly required from the user are bolded in the table below, with the others being filled out by the geometry creation tools.

<b><i>B</i></b>	Header indicating the blade number index. This line is for reference only, and is not used internally in CACTUS.
<b><i>NElem</i></b>	Number of elements.
<b><i>FlipN</i></b>	Set to 1 to flip the element normal direction, 0 otherwise. Nominal element normal direction in CACTUS is calculated to align with the cross product of the blade tangent vector and the quarter-chord line direction.
<b><i>QCx</i></b>	Blade quarter-chord line x coordinates at element ends divided by reference radius (NElem + 1 values).
<b><i>QCy</i></b>	Blade quarter-chord line y coordinates at element ends divided by reference radius (NElem + 1 values).
<b><i>QCz</i></b>	Blade quarter-chord line z coordinates at element ends divided by reference radius (NElem + 1 values).
<b><i>tx</i></b>	Blade unit tangent vector (rearward chord line direction) x component at element ends (NElem + 1 values).
<b><i>ty</i></b>	Blade unit tangent vector (rearward chord line direction) y component at element ends (NElem + 1 values).
<b><i>tz</i></b>	Blade unit tangent vector (rearward chord line direction) z component at element ends (NElem + 1 values).
<b><i>CtoR</i></b>	Blade chord to turbine reference radius ratio at element ends (NElem + 1 values).
<i>E</i>	Element center x coordinates divided by reference radius (NElem values).
<i>E</i>	Element center y coordinates divided by reference radius (NElem values).
<i>E</i>	Element center z coordinates divided by reference radius (NElem values).
<i>E</i>	Element unit tangent vector (rearward chord line direction) x component (NElem values).
<i>E</i>	Element unit tangent vector (rearward chord line direction) y component (NElem values).
<i>E</i>	Element unit tangent vector (rearward chord line direction) z component (NElem values).
<i>E</i>	Element unit normal vector x component (NElem values).
<i>E</i>	Element unit normal vector y component (NElem values).
<i>E</i>	Element unit normal vector z component (NElem values).
<i>E</i>	Element unit spanwise vector x component (NElem values).

<i>E</i>	Element unit spanwise vector y component (NElem values).
<i>E</i>	Element unit spanwise vector z component (NElem values).
<i>EC</i>	Element chord to turbine reference radius ratio (NElem values).
<i>EA</i>	Element area divided by turbine reference radius squared (NElem values).
	Index of the foil section data to be applied to each element (NElem values). This index corresponds to the order in which the foil section data table files are supplied in the namelist input file.

There should be a separate blade data block for each blade in the turbine rotor.

Note that the direction of the element normal vector defines the sign of the flow angle of attack calculated on that element and used to interpolate the foil data tables. Positive angle of attack will be calculated when the relative flow velocity component in the element normal direction is positive. Flow angle of attack is zero when the relative flow velocity is aligned with the element tangent vector.

#### 4.1.1.3 Strut

The strut geometry is specified by the user at the element end points in terms of the locations of the strut mid-chord line, and the local chord-to-radius ratio at the initial time of the turbine simulation. The element specific geometry should be calculated consistent with the element end geometry using the geometry creation tools described in Section 4.1.3. The parameters directly required from the user are bolded in the table below, with the others being filled out by the geometry creation tools.

	Header indicating the strut number index. This line is for reference only, and is not used internally in CACTUS.
<b><i>NElem</i></b>	Number of elements.
<b><i>TtoC</i></b>	Strut thickness to chord ratio (single value, assumed constant over strut).
<b><i>MCx</i></b>	Strut mid-chord x coordinates at element ends divided by reference radius (NElem + 1 values).
<b><i>MCy</i></b>	Strut mid-chord x coordinates y coordinates at element ends divided by reference radius (NElem + 1 values).
<b><i>MCz</i></b>	Strut mid-chord x coordinates z coordinates at element ends divided by reference radius (NElem + 1 values).
<b><i>CtoR</i></b>	Strut chord to turbine reference radius ratio at element ends (NElem + 1 values).
<i>E</i>	Element center x coordinates divided by reference radius (NElem values).
<i>E</i>	Element center y coordinates divided by reference radius (NElem values).
<i>E</i>	Element center z coordinates divided by reference radius (NElem values).
<i>E</i>	Element unit spanwise vector x component (NElem values).
<i>E</i>	Element unit spanwise vector y component (NElem values).



<i>E</i>	Element unit spanwise vector z component (NElem values).
<i>EC</i>	Element chord to turbine reference radius ratio (NElem values).
<i>EA</i>	Element area divided by turbine reference radius squared (NElem values).
<i>BIndS</i>	Index of the blade to which the first strut element connects. Used for blade-strut interference drag calculation. Set to zero if start element connected to rotor shaft.
<i>EIndS</i>	Index of the element on the above blade at which the first strut element connects. Used for blade-strut interference drag calculation. Set to zero if first element connected to rotor shaft.
<i>BIndE</i>	Index of the blade to which the last strut element connects. Used for blade-strut interference drag calculation. Set to zero if last element is connected to rotor shaft.
<i>EIndE</i>	Index of the element on the above blade to which the last strut element connects. Used for blade-strut interference drag calculation. Set to zero if last element is connected to rotor shaft.

There should be a separate strut data block for each strut in the turbine rotor.

#### 4.1.2 File Format Example

A file format example is given below. This example can also be found in  
/ G / A . in the CACTUS repository.

---

```

NBlade: 2
NStrut: 2
RotN: 0.00000e+00 1.00000e+00 0.00000e+00
RotP: 0.00000e+00 0.00000e+00 0.00000e+00
RefAR: 3.52000e+00
RefR: 3.15000e+01
Type: VAWT
Blade 1:
  NElem: 5
  FlipN: 0
  QCx: -1.25936e-02 -1.25936e-02 -1.25936e-02 -1.25936e-02 -1.25936e-02 -1.25936e-02
  QCy: 0.00000e+00 5.28000e-01 1.05600e+00 1.58400e+00 2.11200e+00 2.64000e+00
  QCz: -0.00000e+00 -6.40000e-01 -9.60000e-01 -9.60000e-01 -6.40000e-01 -0.00000e+00
  tx: 1.00000e+00 1.00000e+00 1.00000e+00 1.00000e+00 1.00000e+00 1.00000e+00
  ty: 0.00000e+00 0.00000e+00 0.00000e+00 0.00000e+00 0.00000e+00 0.00000e+00
  tz: 0.00000e+00 0.00000e+00 0.00000e+00 0.00000e+00 0.00000e+00 0.00000e+00
  CtoR: 7.40800e-02 7.40800e-02 7.40800e-02 7.40800e-02 7.40800e-02 7.40800e-02
  PEx: -1.25936e-02 -1.25936e-02 -1.25936e-02 -1.25936e-02 -1.25936e-02
  PEy: 2.64000e-01 7.92000e-01 1.32000e+00 1.84800e+00 2.37600e+00
  PEz: -3.20000e-01 -8.00000e-01 -9.60000e-01 -8.00000e-01 -3.20000e-01
  tEx: 1.00000e+00 1.00000e+00 1.00000e+00 1.00000e+00 1.00000e+00
  tEy: 0.00000e+00 0.00000e+00 0.00000e+00 0.00000e+00 0.00000e+00
  tEz: 0.00000e+00 0.00000e+00 0.00000e+00 0.00000e+00 0.00000e+00
  nEx: -0.00000e+00 -0.00000e+00 0.00000e+00 0.00000e+00 0.00000e+00
  nEy: 7.71373e-01 5.18302e-01 0.00000e+00 -5.18302e-01 -7.71373e-01
  nEz: 6.36383e-01 8.55198e-01 1.00000e+00 8.55198e-01 6.36383e-01
  sEx: -0.00000e+00 -0.00000e+00 -0.00000e+00 -0.00000e+00 -0.00000e+00
  sEy: -6.36383e-01 -8.55198e-01 -1.00000e+00 -8.55198e-01 -6.36383e-01
  sEz: 7.71373e-01 5.18302e-01 -0.00000e+00 -5.18302e-01 -7.71373e-01

```

ECtoR: 7.40800e-02 7.40800e-02 7.40800e-02 7.40800e-02 7.40800e-02  
EAreaR: 6.14634e-02 4.57371e-02 3.91142e-02 4.57371e-02 6.14634e-02  
iSect: 1 1 1 1 1

Blade 2:

NElem: 5  
FlipN: 0  
QCx: 1.25936e-02 1.25936e-02 1.25936e-02 1.25936e-02 1.25936e-02 1.25936e-02  
QCy: 0.00000e+00 5.28000e-01 1.05600e+00 1.58400e+00 2.11200e+00 2.64000e+00  
QCz: 1.54227e-18 6.40000e-01 9.60000e-01 9.60000e-01 6.40000e-01 1.54227e-18  
tx: -1.00000e+00 -1.00000e+00 -1.00000e+00 -1.00000e+00 -1.00000e+00 -1.00000e+00  
ty: 0.00000e+00 0.00000e+00 0.00000e+00 0.00000e+00 0.00000e+00 0.00000e+00  
tz: -1.22465e-16 -1.22465e-16 -1.22465e-16 -1.22465e-16 -1.22465e-16 -1.22465e-16  
CtoR: 7.40800e-02 7.40800e-02 7.40800e-02 7.40800e-02 7.40800e-02 7.40800e-02  
PEx: 1.25936e-02 1.25936e-02 1.25936e-02 1.25936e-02 1.25936e-02  
PEy: 2.64000e-01 7.92000e-01 1.32000e+00 1.84800e+00 2.37600e+00  
PEz: 3.20000e-01 8.00000e-01 9.60000e-01 8.00000e-01 3.20000e-01  
tEx: -1.00000e+00 -1.00000e+00 -1.00000e+00 -1.00000e+00 -1.00000e+00  
tEy: 2.41486e-19 -9.83382e-19 0.00000e+00 9.83382e-19 -2.41486e-19  
tEz: -1.22172e-16 -1.23061e-16 -1.22465e-16 -1.23061e-16 -1.22172e-16  
nEx: 7.79344e-17 1.04732e-16 1.22465e-16 1.04732e-16 7.79344e-17  
nEy: 7.71373e-01 5.18302e-01 0.00000e+00 -5.18302e-01 -7.71373e-01  
nEz: -6.36383e-01 -8.55198e-01 -1.00000e+00 -8.55198e-01 -6.36383e-01  
sEx: 9.40865e-17 6.46235e-17 -0.00000e+00 -6.46235e-17 -9.40865e-17  
sEy: -6.36383e-01 -8.55198e-01 -1.00000e+00 -8.55198e-01 -6.36383e-01  
sEz: -7.71373e-01 -5.18302e-01 -0.00000e+00 5.18302e-01 7.71373e-01  
ECtoR: 7.40800e-02 7.40800e-02 7.40800e-02 7.40800e-02 7.40800e-02  
EAreaR: 6.14634e-02 4.57371e-02 3.91142e-02 4.57371e-02 6.14634e-02  
iSect: 1 1 1 1 1

Strut 1:

NElem: 5  
TtoC: 1.50000e-01  
MCx: 0.00000e+00 0.00000e+00 0.00000e+00 0.00000e+00 0.00000e+00 0.00000e+00  
MCy: 1.32000e+00 1.32000e+00 1.32000e+00 1.32000e+00 1.32000e+00 1.32000e+00  
MCz: -0.00000e+00 -1.92000e-01 -3.84000e-01 -5.76000e-01 -7.68000e-01 -9.60000e-01  
CtoR: 7.40800e-02 7.40800e-02 7.40800e-02 7.40800e-02 7.40800e-02 7.40800e-02  
PEx: 0.00000e+00 0.00000e+00 0.00000e+00 0.00000e+00 0.00000e+00  
PEy: 1.32000e+00 1.32000e+00 1.32000e+00 1.32000e+00 1.32000e+00  
PEz: -9.60000e-02 -2.88000e-01 -4.80000e-01 -6.72000e-01 -8.64000e-01  
sEx: 0.00000e+00 0.00000e+00 0.00000e+00 0.00000e+00 0.00000e+00  
sEy: 0.00000e+00 0.00000e+00 0.00000e+00 0.00000e+00 0.00000e+00  
sEz: -1.00000e+00 -1.00000e+00 -1.00000e+00 -1.00000e+00 -1.00000e+00  
ECtoR: 7.40800e-02 7.40800e-02 7.40800e-02 7.40800e-02 7.40800e-02  
EAreaR: 1.42234e-02 1.42234e-02 1.42234e-02 1.42234e-02 1.42234e-02  
BIndS: 0  
EIndS: 0  
BIndE: 1  
EIndE: 3

Strut 2:

NElem: 5  
TtoC: 1.50000e-01  
MCx: 0.00000e+00 -2.35132e-17 -4.70264e-17 -7.05397e-17 -9.40529e-17 -1.17566e-16  
MCy: 1.32000e+00 1.32000e+00 1.32000e+00 1.32000e+00 1.32000e+00 1.32000e+00  
MCz: 0.00000e+00 1.92000e-01 3.84000e-01 5.76000e-01 7.68000e-01 9.60000e-01  
CtoR: 7.40800e-02 7.40800e-02 7.40800e-02 7.40800e-02 7.40800e-02 7.40800e-02  
PEx: -1.17566e-17 -3.52698e-17 -5.87830e-17 -8.22963e-17 -1.05809e-16  
PEy: 1.32000e+00 1.32000e+00 1.32000e+00 1.32000e+00 1.32000e+00

```

PEz: 9.60000e-02 2.88000e-01 4.80000e-01 6.72000e-01 8.64000e-01
sEx: -1.22465e-16 -1.22465e-16 -1.22465e-16 -1.22465e-16 -1.22465e-16
sEy: 0.00000e+00 0.00000e+00 0.00000e+00 0.00000e+00 0.00000e+00
sEz: 1.00000e+00 1.00000e+00 1.00000e+00 1.00000e+00 1.00000e+00
ECtoR: 7.40800e-02 7.40800e-02 7.40800e-02 7.40800e-02 7.40800e-02
EAreaR: 1.42234e-02 1.42234e-02 1.42234e-02 1.42234e-02 1.42234e-02
BIndS: 0
EIndS: 0
BIndE: 2
EIndE: 3

```

---

### 4.1.3 Geometry Creation Tools

A set of MATLAB tools have been created to allow the user to generate a CACTUS geometry input file for an arbitrary turbine rotor. This set of MATLAB scripts is located in the `C G` folder in the CACTUS source code directory. Note that these scripts should also run without modification under GNU Octave (free software package that mimics much of the basic MATLAB syntax), with the possible exception of the plotting functions in the script `G .`

The script `C .` generates an empty turbine geometry structure and optionally fills it out with data for a parameterized generic turbine type. See the comments at the top of the file for details. The user can then modify the data in the created geometry structure as necessary to represent their particular problem. When defining blade and strut geometry, the user should first fill out the fields indicated in bold in Sections 4.1.1.2 and 4.1.1.3, and then use the scripts `C BEG .` and `C EG .` to fill out the blade and strut element geometry fields.

Once the turbine geometry structure has been finalized, the script `G .` will create the CACTUS geometry input file. The script `G .` will read an existing CACTUS geometry input file and create a corresponding geometry structure in MATLAB for further modification.

The script `G .` will plot the geometry contained in a turbine geometry structure at a particular phase angle of rotation around the turbine rotation axis. The plotting functions in this script have not been verified to work in GNU Octave.

An example script that creates and plots turbine rotor geometry for a vertical axis wind turbine is provided in Appendix A.

## 4.2 Foil Table Data

Foil data table files contain the force coefficient data and dynamic stall parameters for a foil to be used in the calculation. The user can specify multiple foil data table files in the CACTUS namelist input file and they are indexed in the order they appear in the namelist input file. The `foil` parameter in the geometry input file identifies the index of the foil data table to be applied to each blade element section.

The foil coefficients are input as lift, drag, and pitching moment (about the 25% chord point) coefficients (per span) as a function of angle of attack (AOA) and Reynolds number. A maximum of 20 Reynolds number values are allowed. A maximum of 1000 AOA values for each Reynolds number are allowed. The AOA data must go from -180 to 180 deg. The sign convention for AOA is such that positive AOA on a blade element is generated by positive relative flow velocity in the positive element normal direction. Note that the cross-flow turbine geometry generator provided with CACTUS creates element normal vectors in the machine inward direction. The axial flow turbine geometry generator creates element normal vectors in the machine rearward (downwind) direction at zero blade incidence angle.

Parameter definitions and a file format example are below. Note that a number of foil data table files are provided in the *A* *D* folder in the CACTUS source code directory.

## 4.2.1 File Parameters

### 4.2.1.1 Header

	Title for this foil data table. For reference only; not used internally in CACTUS.
$C$	Thickness to chord ratio for this foil section.
$L \quad A \quad A$	Angle of attack (deg) at zero lift for this section.
$C \quad D$	This flag allows the user to reverse the orientation of a non-symmetric foil with respect to the normal vector of the blade element section to which it is applied (set the flag to 1).

### 4.2.1.2 Reynolds Number Data Block

$B \quad D \quad .$	The Boeing-Vertol dynamic stall model uses reference angle of attack values to switch from a steady attached flow state to a dynamic stalled state. While these are denoted as "stall" AOA values, they should be set back from the foil stall AOA such that the lift coefficient is still fairly linear with AOA at this point. Generally, a value 50 - 75% of the way between the zero lift AOA and the stall AOA works well.
$LB \quad D \quad .$	The Leishman-Beddoes dynamic stall model uses a reference lift slope and a critical lift coefficient value to indicate the onset of leading edge stall. The critical lift coefficient value should be approximately equal to the value of lift coefficient that would have been obtained at the foil stall AOA, had the lift coefficient remained linear with AOA (with slope given by the reference lift coefficient slope).

$F$	$C$	Foil force and moment coefficient data at angle of attack from -180 to 180 deg. (AOA, lift coefficient, drag coefficient, pitching moment coefficient about the 25% chord point).
-----	-----	---

There should be one Reynolds number data block for each Reynolds number at which foil data exists.

### 4.2.2 File Format Example

A file format example is given below.

---

Title: AFTitle

Thickness to Chord Ratio: 0.2

Zero Lift AOA (deg): 0.0

Reverse Camber Direction: 0

Reynolds Number: 1e6

BV Dyn. Stall Model - Positive Stall AOA (deg): 10

BV Dyn. Stall Model - Negative Stall AOA (deg): -10

LB Dyn. Stall Model - Lift Coeff. Slope at Zero Lift AOA (per radian): 6.28

LB Dyn. Stall Model - Positive Critical Lift Coeff.: 1.3

LB Dyn. Stall Model - Negative Critical Lift Coeff.: -1.3

AOA (deg) CL CD Cm25

-180.0 0.0 1.0 0.0

... ..

180.0 0.0 1.0 0.0

Reynolds Number: 5e6

...

---

## 4.3 CACTUS Namelist Input

This section describes the FORTRAN namelist input file for CACTUS. There are two namelist groups in the input file, `&C I` and `&C I`. The parameters that can be input in each group are given in the tables below. Parameters not specified in the namelist input are left at the indicated default values. The parameters listed in bold font below are the parameters most commonly specified in an input file (with the rest being left at default values).

### 4.3.1 Configuration Inputs

Regression Testing	
$F$	Set to 1 to perform a regression test (two iterations, generates RegData.out output file), 0 for normal operation (default).

<b>Wall calculation</b>	
$G \ F$	Set to 1 to use a ground plane, 0 otherwise (default 0).
$F \ F$	Set to 1 to use a free surface, 0 otherwise (default 0).
$G \ G \ F$	Factor on default ground plane grid spacing (default 1).
$F \ G \ F$	Factor on default free surface near-field grid spacing (default 1).
<b>Calculation inputs</b>	
<b><math>nr</math></b>	Number of revolutions to perform (default 10)
<b><math>nti</math></b>	Number of time steps per revolution (default 20)
<b><math>convr</math></b>	Convergence level for the revolution average power coefficient. Iteration will finish before    revs if this level is hit. Input -1 to skip convergence check (default).
<b><math>iut</math></b>	Number of iterations between wake convection velocity updates. If set to zero, the interval will be calculated automatically. If negative, wake convection velocities will be left at the values calculated at the time the wake element is created (no wake convection velocity updates).
<b><math>iWall</math></b>	Number of iterations between wall model updates (if wall calculation is active).
$F \ F$	Flag to enable timestep filtering. Set to 1 to enable filtering of the blade bound vorticity smooth over    timesteps (often needed for stability when blade chord to radius ratio is high). Set to 0 for no filtering (default).
	Number of timesteps over which the blade bound vorticity is filtered smooth when $F \ F = 1$ (default 3).
	Flag to specify the finite vortex core model to use. Input 1 for constant velocity in the core (default). Input 2 to use linear radial velocity distribution in core. Input 0 to turn off core model.
	Factor on nominal bound vortex core radius used if    = 1 (default 1). Nominal bound vortex core radius is specified by the maximum blade chord value input in the turbine geometry specification.
	Factor on nominal trailing wake vortex core radius used if    = 1 (default 1). Nominal trailing wake vortex core radius is specified by the maximum blade element span value input in the turbine geometry specification.
	Factor on nominal spanwise wake vortex core radius used if    = 1 (default 1). Nominal spanwise wake vortex core radius is calculated from a reference distance between spanwise wake lines given the temporal discretization level used.
<b><math>I</math></b>	1 to ignore any compressibility effects in models, 0 to include compressibility effects (default).
	1 to use final convergence step, 0 to not (default). If selected, the temporal discretization level is refined once/if initial convergence is reached before    revolutions have been performed.
	Revolution number after which to switch to final convergence, if initial convergence level has not yet been achieved. Input -1 to skip this

	check (default).
	Final number of time steps per revolution. This value will replace during final convergence. Input -1 to leave = (default).
	Final convergence level. This level will replace convrg during final convergence. Input -1 to skip final convergence check (default).
	Final number of iterations between wake updates. This value will replace during final convergence. Default behavior is the same as .
	1 to ignore wake points beyond $x/R =$ , 0 to use all wake points (default).
	If ixterm = 1, defines $x/R$ beyond which wake points are ignored (default 5)
<b>Dynamic aero effects</b>	
<i>DSFlag</i>	0 for no dynamic stall, 1 for Modified Boeing-Vertol model (default), 2 for Leishman-Beddoes model.
<i>F</i>	0 for no element pitch rate aerodynamic effects, 1 to include these effects (default).
<b>Optional outputs</b>	
<i>Output_ELFlag</i>	1 to output full detail element loads .csv file, 0 to omit this output (default)
<i>WallOutFlag</i>	1 to output wall panel data (averaged over the last revolution), 0 to omit this output (default).
<i>DiagOutFlag</i>	1 to output diagnostic info to standard output device each iteration, 0 to omit this output (default).

### 4.3.2 Case Inputs

<i>Jbtitle</i>	(string in single quotes) Job title.
<b>Operation point inputs</b>	
<i>RPM</i>	Rotor rotation rate (revs per minute).
<i>Ut</i>	Tip speed ratio with freestream flow speed ( $U_{tip}/U_{inf}$ )
<i>rho</i>	Density (slugs/ft <sup>3</sup> )
<i>vis</i>	Dynamic viscosity (slugs/(ft*s))
<i>tempr</i>	Temperature (degF)
<i>hBLRef</i>	Height above ground of the effective freestream to be used in ground shear layer model (ft).
<i>slex</i>	Exponent for ground shear layer model (Ex. 1/2 for parabolic laminar BL model, 1/7 turbulent approx., 0 for constant freestream).

<b><i>hAG</i></b>	Height above ground at turbine geometry origin point (ft). Note that the ground plane is assumed to be oriented with its normal vector in the +y direction.
<b><i>dFS</i></b>	Depth below un-deflected free surface of the turbine geometry origin point (ft). Only used if free surface calculation is active. Note that the un-deflected free surface is assumed to be oriented with its normal vector in the +y direction.
	1 to activate sinusoidal gust model, 0 to use nominal constant freestream model (default). The sinusoidal gust perturbation is modeled per the IEC 61400-1 Wind Turbine Design Standard.
	Amplitude of sinusoidal gust perturbation (m/s) used if = 1
	Timescale of sinusoidal gust perturbation (s) used if = 1
<b><i>0</i></b>	Starting x location of the gust divided by reference radius (location where perturbation is zero at initial simulation time). Used if = 1.
	1 to activate the tower wake model, 0 otherwise (default). This model adds an empirically defined viscous wake deficit to the nominal freestream to model the presence of the support tower. The model currently assumes this tower to be oriented along the y-axis.
	Number of elements used to represent the tower (default 10). Used if = 1.
	Tower location x coordinate divided by reference radius. Used if = 1.
	Lower tower y coordinate divided by reference radius. Used if = 1.
	Upper tower y coordinate divided by reference radius. Used if = 1.
<b><i>D</i></b>	Tower diameter divided by reference radius. Used if = 1.
<b><i>CD</i></b>	Tower 2D sectional drag coefficient based on tower diameter (default 1.0). Used if = 1.
<b>Geometry file</b>	
<b><i>GeomFilePath</i></b>	(string in single quotes) Path to turbine geometry input file.
<b>Airfoil section data</b>	
<b><i>nSect</i></b>	Number of airfoil section data tables to use (default 1)
<b><i>AFDPath</i></b>	(string in single quotes) Array (comma separated) of section data file path strings (size=nsect).
<b>Other parameters</b>	
<b><i>CD</i></b>	Additional parasitic interference drag coefficient based on "chord area" (chord squared) to be applied to the blade/strut interference drag calculation (default 0)



$C_E$	Additional machine level excrescence torque coefficient, $C_{T,ExcM} = T_{ExcM} / \left( \frac{1}{2} \rho U_{Tip}^2 R^3 \right) \cdot (\text{default } 0)$
-------	---

### 4.3.3 File Format Example

A file format example is given below. This example can also be found in `/C2/A` in the CACTUS repository.

---

&ConfigInputs

GPFlag = 0

nr = 10

nti = 16

convrg = .0001

iut = 0

ifc = 0

ixterm = 0

ntif = 16

iutf = 1

nrhc = 9

convrgf = .0001

DiagOutFlag=1

Output\_ELFlag=1

/End

&CaseInputs

jbtitle = 'Test VAWT'

rho = .002378

vis = .3739E-6

tempr = 60.0

hBLRef = 56.57

slex = 0.0

hAG = 15.0

---

```

RPM    = 52.0
Ut      = 5.0

! Turbine geometry
GeomFilePath='../TestGeom/TestVAWT.geom'

! Airfoil section data
nSect   = 1
AFDPath = '../Airfoil_Section_Data/NACA_0015.dat'

/End

```

---

## 5 Output Description

This section describes the output files written by CACTUS.

### 5.1 Revolution Average Performance Data

Revolution averaged performance data for each revolution are written to a comma delimited file appended with  $D_{rev}$ .

	Revolution number
$C_p$ (-)	Revolution average machine power coefficient
$C_p^*$ (-)	Revolution average machine power coefficient normalized with $U_{Tip}$ instead of $U_\infty$
$C_t$ (-)	Revolution average torque coefficient
$F_x C_f$ (-)	Revolution average x component of force coefficient
$F_y C_f$ (-)	Revolution average y component of force coefficient
$F_z C_f$ (-)	Revolution average z component of force coefficient
$P$ ( )	Revolution average machine power
$T$ ( - )	Revolution average machine torque

### 5.2 Temporal Performance Data

Performance data for each time step are written to a comma delimited file appended with  $D_{time}$ .

$t_N$ (-)	Normalized simulation time $t_N = t \frac{U_\infty}{R}$
$\theta$ ( )	Turbine rotational phase angle
$C_t$ (-)	Revolution number Torque coefficient

$C$ (-)	Power coefficient
$F C_x$ (-)	X component of force coefficient
$F C_y$ (-)	Y component of force coefficient
$F C_z$ (-)	Z component of force coefficient
The following are replicated for each blade	
$B F C_x$ (-)	Contribution to x component of force coefficient from blade
$B F C_y$ (-)	Contribution to y component of force coefficient from blade
$B F C_z$ (-)	Contribution to z component of force coefficient from blade
$B C$ (-)	Contribution to torque coefficient from blade
The following are replicated for each strut	
$F C_x$ (-)	Contribution to x component of force coefficient from strut
$F C_y$ (-)	Contribution to y component of force coefficient from strut
$F C_z$ (-)	Contribution to z component of force coefficient from strut
$C$ (-)	Contribution to torque coefficient from strut

### 5.3 Temporal Element Loads Data

When  $ELF$  is set to 1 in the namelist input file, element loads data for each time step are written to a comma delimited file appended with  $E D$ .

(-)	Normalized simulation time $t_N = t \frac{U_\infty}{R}$
( )	Turbine rotational phase angle
$B$	Blade number
$E$	Element number
	Revolution number
$A_{A25}$ ( )	Local flow angle of attack, defined at element quarter-chord location.
$A_{A50}$ ( )	Reference 50% chord flow angle of attack. Different from $A_{A25}$ when element is rotating in the local spanwise direction.
$A_{A75}$ ( )	Reference 75% chord flow angle of attack. Different from $A_{A25}$ when element is rotating in the local spanwise direction.
$A$ (-)	Normalized AOA rate $\dot{\alpha}_{Norm} = \frac{\dot{\alpha}_c}{2U_{Loc}}$
(-)	Element Reynolds number based on element chord

$(-)$	Element Mach number
$(-)$	Local flow speed ratio with freestream $U_r = U_{Loc}/U_\infty$
$CL (-)$	Element lift coefficient, $C_L = L / \left( \frac{1}{2} \rho U_{Loc}^2 A_E \right)$
$CD (-)$	Element drag coefficient, $C_D = D / \left( \frac{1}{2} \rho U_{Loc}^2 A_E \right)$
$C_{M,25} (-)$	Element pitching moment coefficient about the quarter-chord location, $C_{M,25} = M_{25} / \left( \frac{1}{2} \rho U_{Loc}^2 A_E c \right)$
$CLC (-)$	Circulatory component of element lift coefficient, $C_{L,Circ} = L_{Circ} / \left( \frac{1}{2} \rho U_{Loc}^2 A_E \right)$
$C_N (-)$	Element normal force coefficient, $C_N = N / \left( \frac{1}{2} \rho U_{Loc}^2 A_E \right)$
$C_T (-)$	Element tangential force coefficient, $C_T = T / \left( \frac{1}{2} \rho U_{Loc}^2 A_E \right)$
$F_x (-)$	Contribution to x component of force coefficient from element
$F_y (-)$	Contribution to y component of force coefficient from element
$F_z (-)$	Contribution to z component of force coefficient from element
$(-)$	Contribution to torque coefficient from element

## 5.4 Wall Model Data

When a wall calculation is being performed and  $F_{wall}$  is set to 1 in the namelist input file, summary output data for the wall calculation is written to a comma delimited file appended with either  $G.D$  for a ground plane calculation, or  $F.D$  for a free surface calculation.

### 5.4.1 Ground Plane Data

$X / (-)$	X location of the panel center normalized by R
$Y / (-)$	Y location of the panel center normalized by R
$Z / (-)$	Z location of the panel center normalized by R
$D / (-)$	Source density on the panel normalized by $U_\infty$

### 5.4.2 Free Surface Data

$X / (-)$	X location of the panel center normalized by R
-----------	--

$/ \quad (-)$	Y location of the panel center normalized by R
$/ \quad (-)$	Z location of the panel center normalized by R
$/ \quad (-)$	Wall tangential velocity (nominal freestream direction) normalized by $U_\infty$
$H/ \quad (-)$	Free surface height (above un-deflected height) normalized by R

## 6 References

1. Murray, J., Barone, M., “The Development of CACTUS, a Wind and Marine Turbine Performance Simulation Code,” AIAA Paper 2011-147, 2011.
2. Strickland, J. H., Smith, T., Sun, K., “A Vortex Model of the Darrieus Turbine: An Analytical and Experimental Study,” Sandia National Laboratories, SAND81-7017, Albuquerque, NM, 1981.
3. Katz, J., Plotkin, A., Low Speed Aerodynamics, 2nd ed., Cambridge University Press, Cambridge, England, UK, 2001.

# Appendix A: Example Geometry Script

Below is an example geometry creation MATLAB script for a vertical axis wind turbine. This example can also be found in `/G/A` in the CACTUS repository.

---

```
%
%
% A
% A
% ( , '.../.../ ' );
%
%
% =31.5;
% =2.64;
% =0.07408;
% =.42;
%
%
% =2;
% =5;
% =2;
% =5;
%
% =
% =.15;
%
%
% = ' A . ' ;
%
% ?
% =1;
%
%
% = /180;
%
%
% = ' A ' ;
% =1;
%
% = ( , , , , , , , , , 1, , , ,
% , , );
%
%
% ( , );
%
%
%
%
% = -4,4 ;
% = -2,4 ;
% = -4,4 ;
```

⌘

```

=1;
=.5;
=.5;

=
(1);
( , ' ' , 303 124 956 610 )
( , ' ' , 5.2743 -002 5.1245 -002 8.9979 -001 8.8141 -001 )
( , ' ' , -52.1999 30.4749 62.2119 )
( , ' ' , 1.8643 -001 9.7433 -001 -1.2615 -001 )
( , ' ' A , 6.3060 +000)

( , ' ' ' ' ' ');
=
( ' ' , -1,0,0 );
( , ' ' ' ' ' ');
( , ' ' A , 1,1,1 )
( , ' ' ' , ' , ' , ' , ' )

= ;
=
(0,2* ,150);
=1: ( )
=
( , , ( ) , , , , );
= ;
(.01);

```