

LUCERNE UNIVERSITY OF APPLIED SCIENCE & ARCHITECTURE

ARIS - DATA FUSION FOR A SOUNDING ROCKET

BACHELOR THESIS



Author:	Michael Kurmann
Supervisor:	Prof. Marcel Joss
Expert:	Werner Scheidegger
Industrial Partner:	ARIS (Akademische Raumfahrt Initiative Schweiz) Oliver Kirchhoff
Submission date:	22.12.2017
Classification:	Access

Declaration

Hereby, I declare that I have composed the presented paper independently on my own and without any other resources than the ones indicated. All thoughts taken directly or indirectly from sources are properly denoted as such. This paper has neither been previously submitted to another authority nor has it been published yet.

Horw, March 8, 2018

Abstract

This is the Abstract

Contents

1	Introduction	2
1.1	Bla	2
1.2	Research	2
1.3	Problem	2
1.4	Requirements	3
1.5	Desired Solution	3
2	Approach	4
3	Tests	5
4	Conclusion	6
4.1	Thanks	6
	Appendices	8

Chapter 1

Introduction

Simon (2006)

1.1 Bla

About Sensor Fusion and ARIS

1.2 Research

Write about the Papers/book you used: Kalman-filter Optimal state estimation The Master Thesis

1.3 Problem

State the problem, what will be difficult ? For what is it ? where should it be improved

Problems found so far:

- How to calculate Height out of Pressure/Temp/Humidity Fabian ver: $44330 * (1 - (\frac{pressure}{101325})^{\frac{1}{5.255}})$
- How to integrate AirBreaks/Drag Force of Air/ Trust of Motor ?
- What are the different noise factors and when do they occur ?
- The up-flight is rather short: about 25s so the Fusion should have a small settling time
- The Chip on which it is used is no the fastest : 168 MHz clock
- The Ram on the Chip is not endless: Maximal space for the Sensor fusion is about 10kB
- The Sensor Fusion should be as modular as possible so that it also can be used in the next competition
- The Sensor Fusion has to be as sturdy as possible so that it will not fail if a problem occurs
- The Fusion should make a state Estimation as precise as possible.
- There are a lot of different variables: 3 Positions, 1 Speed, 3 Accelerations, 3 Lagen, Time, Pressure, Temperature, Humidity, Up-/Downforce.
- Especially the Input Value u which is the force onto the rocket is difficult to define (Drag, Trust = acceleration depends on weight which changes over time).
- The different Sensor have different weaknesses:
 - Accelerometer: Offset, drift, weak to vibrations
 - Gyro: Weak to Vibrations
 - Barometer: Many uncertainties, unprecise
 - GPS: Slow (max 5Hz)

Requirement	Rating	Aim	Importance
Precision	Error between estimation and ground truth	< 5% after settling time	High
Sturdiness	% Precision without # sensors	-10% per failed sensor	Medium
Code Size	kB of RAM needed	< 10kB size	High
System Load	Time per loop on the current Chip (128 MHz)	< 1ms	Critical
Non Linearity	# non linearities in the algorithm	0	Desirable
Settling time	Time from ignition to optimal estimation	< 5s	Critical
Modularity	Effort needed to integrate a new sensor	< 10h work	Desirable

Table 1.1: Requirements table

1.4 Requirements

These are the requirements which were drawn out of the problems list.

1.5 Desired Solution

Describe in quick terms what you are aiming for.

Chapter 2

Approach

hoho

Chapter 3

Tests

Here come the tests

Chapter 4

Conclusion

4.1 Thanks

Thanks to the Aris Team espacially Thomas and Fabian Also Thanks to Lukas and Jossely

Bibliography

Simon, D. (2006). *Optimal state estimation : Kalman, H_∞ , and nonlinear approaches*.

Appendices

This is the Appendix