

Aerostructures Exam II Formula Sheet

1 Beam Columns

Granddaddy Equation of Beam Buckling:

$$P_{cr} = \frac{\pi^2 EI}{KL^2} \quad (1)$$

where

- P_{cr} , Euler Critical Load,
- E , Modulus of Elasticity,
- K , Effective Length Factor,
- L , Column Length.

2 Crippling Stress

2.1 Gerard Method

The Gerard method has two equations, those that apply to Channel and Zee sections, and those that apply to all other sections. For all other sections, you must determine the number of flanges and cuts, the g value. Equation 2 is used for Channel and Zee sections while Equation 3 is used for all other sections. There is a table that gives the g values for certain shapes.

$$F_{cc} = B \left(\frac{1}{\frac{A}{t^2}} \right)^m \sqrt[3]{E^m F_{cy}^{(3-m)}} \quad (2)$$

where $B = 4.05$ and $m = 0.82$.

$$F_{cc} = B \left(\frac{1}{\frac{A}{gt^2}} \right)^m \sqrt{E^m F_{cy}^{(2-m)}} \quad (3)$$

where $B = 0.58$ and $m = 0.8$

2.2 Bruhn Method

Divide the shape into different sections and calculate the individual stresses σ using Equation 4. If you get a value above the yield strength of the material σ_y , the allowable stress is then the yield stress. K_c is the buckling coefficient, which will have to be obtained from a graph. Once you have calculated the individual stresses, calculate the total force and divide by the total area to get the average stress (Equation 5)

$$\sigma = \frac{K_c \pi^2 E}{12(1 - \nu_e^2)} \left(\frac{t}{b} \right)^2 \quad (4)$$

$$\sigma_{avg} = \frac{\sum \sigma_i A_i}{\sum A_i} \quad (5)$$

For aluminum sections, we can use the following equations to calculate σ , one for free at one end (Equation 6) and simply supported at the other (Equation 7), and another for simply supported at both ends.

$$\sigma = 0.452E \left(\frac{t}{b} \right)^2 \quad (6)$$

$$\sigma = 3.62E \left(\frac{t}{b} \right)^2 \quad (7)$$

2.2.1 Correction For Plasticity

Once σ_{avg} is calculated, divide it by $F_{0.7}$ and use the Plasticity Correction graph from Bruhn to determine the corrected Crippling Stress. Once the corrected crippling stress is determined, total allowable load can be calculated by multiplying the allowable stress by the area.

3 Flexural Instability

Crippling will cause a beam to fail before Granddaddy Euler Equation 1 predicts. Therefore, to take into account crippling, use Equation 8. Remember that ρ is $\sqrt{\frac{I}{A}}$, or the radius of gyration. The c value depends on the type of rivets that are being used; default to 1.

$$F_c = F_{cc} \left[1 - \frac{F_{cc} \left(\frac{l}{\rho\sqrt{c}} \right)^2}{4\pi^2 E} \right] \quad (8)$$

4 Inter-Rivet Buckling

The skin between the rivets will buckle and to prevent that, you should space the rivets closer together. If you know the flexural instability of the object, F_c , then you can set that to the inter-rivet buckling stress, because you are screwed if you exceed that value. To calculate inter-rivet pitch, use Equation 9

$$s = t \sqrt{\frac{0.9cE}{F_{ir}}} \quad (9)$$