

# Energy release rate of the fiber/matrix interface crack in UD composites under transverse loading: debond-debond and debond-free boundary interactions

Luca Di Stasio<sup>a,b</sup>, Janis Varna<sup>b</sup>, Zoubir Ayadi<sup>a</sup>

<sup>a</sup> *Université de Lorraine, EEIGM, IJL, 6 Rue Bastien Lepage, F-54010 Nancy, France*

<sup>b</sup> *Luleå University of Technology, University Campus, SE-97187 Luleå, Sweden*

---

## Abstract

The effects of crack shielding, finite thickness of the composite and fiber content on fiber/matrix debond growth in thin unidirectional composites are investigated analyzing Representative Volume Elements (RVEs) of different ordered microstructures. Debond growth is characterized by estimation of the Energy Release Rates (ERRs) in Mode I and Mode II using the Virtual Crack Closure Technique (VCCT) and the J-integral. It is found that increasing fiber content, larger distance between debonds in the loading direction and the presence of a free surface close to the debond have all a strong enhancing effect on the ERR. The presence of fully bonded fibers in the composite thickness direction has instead a constraining effect, and it is shown to be very localized. An explanation of these observations is proposed based on mechanical considerations.

*Keywords:* Polymer-matrix Composites (PMCs), Thin-ply, Transverse Failure, Debonding, Finite Element Analysis (FEA)

---

## 1. Introduction

Stimulated by the ever more stringent requirements in terms of weight and mechanical performances of the aerospace industry, in recent years the composite community has returned its attention to the mechanisms of intralaminar crack initiation in general and to initiation of multiple cracking in thin-ply laminates in particular. Alternative design approaches are now considered based on

this non-conventional laminate in applications ranging from cryogenic pressure vessels [1], to airplanes' wings [2], and even reusable space launchers [3].

*Thin-ply* laminates are the result of a technological innovation, the *spread tow* 10 *technology*, which consists in opening or spreading the tows in which fibers (carbon, glass, aramid, basalt among others) are usually shipped in into very thin tapes then used for laminate production. Ply thicknesses of less than  $50\ \mu m$  can nowadays be mass-produced, and record thicknesses of around  $20 - 25\ \mu m$ , or  $\sim 4 - 5$  times the average fiber's diameter, have been achieved. The technique 15 in its current form, sometimes referred to as "FUKUI method" from the name of the Japanese prefecture it originated in, was firstly proposed towards the end of the 1990s [4] and perfected in the subsequent decade [5, 6].

Several experimental investigations on *thin ply* laminates have highlighted their main properties [7, 8, 9, 10, 11, 12, 13, 14, 15, 16, 17, 18, 19]: increased fiber con- 20 tent; more uniform packing of fibers; delay and even suppression of intralaminar cracking (called also transverse-, matrix- or micro-cracking) and delamination. A very insightful work documenting how these mechanisms are affected by the morphology of *thin ply* laminates is the microscopic study of Saito & al. [20], which focuses on the effect of ply thickness on the onset and propagation of 25 intralaminar cracking. In their investigation, tensile tests were performed on carbon fiber/epoxy  $[0_2, 90_n, 0_2]$  *thin-ply* laminates for  $n = 1, 2, 4$  and the crack density was measured at several levels of applied tensile strain in the range between 0% and 1.5%. Furthermore, they performed microscopic observations on the specimen's edge at each level of strain. They observed the onset of 30 fiber/matrix interface cracks (referred to as debonds in the following) at lower levels of strain in thinner plies, while at the same time coalescence of debonds and through-the-thickness propagation of transverse cracks in thin plies were delayed and even suppressed as ply thickness decreased. In particular, they reported the first onset of debonds at 0.4% for  $n = 1, 2$  and 0.7% for  $n = 4$ . For 35  $n = 1$ , however, at  $\varepsilon = 1.5\%$  coalescence of debonds had started to take place but the crack had not completely propagated through the thickness, while for  $n = 2$  and  $n = 4$  the latter already happened at a value of strain respectively

of 1.3% and 1%. Our inability to explain these observations with the currently accumulated knowledge demonstrates the necessity of further investigation of  
40 interactions between debonds and studies of the constraining (or accelerating) effect of presence of bonded fibers, free and constrained boundaries in the vicinity of a partially debonded fiber.

Early studies on the effect of ply thickness on the onset and propagation of transverse cracks were conducted on glass fiber/epoxy cross-ply laminates by Bailey,  
45 Parvizi and collaborators [21, 22, 23], who firstly observed the beneficial effect of thickness reduction on the delay of transverse cracking. They furthermore pointed the attention to the appearance of debonds at the fiber/matrix interface and their subsequent coalescence as the mechanism at the origin of transverse cracks [24]. Moreover, they identified the main mechanical driver of the damage  
50 process in the mismatch of elastic properties, and particularly of Poisson's ratios, between fibers and matrix [25]. A full understanding of damage onset and propagation in *thin-ply* laminates thus requires the comprehension of the mechanisms governing its very first stage, i.e. the fiber/matrix interface crack. First results were obtained through analytical models in the case of a single fiber with  
55 an arc crack (debond) in an infinite matrix under transverse tension by England [26] and Perlman & Sih [27], who obtained the stresses at the interface and calculated the stress intensity factors at the crack tip, and by Toya [28], who evaluated the Energy Release Rate (ERR). Drawing upon the results for the straight bi-material interface crack by Comninou [29], the effect of crack face contact in  
60 fiber-matrix debonding was investigated in [30, 31]. In [32], it was showed in terms of ERR why the case of a single asymmetric debond is more likely to be observed under remote transverse tension than two symmetric debonds on the same fiber. The effect of different types and combinations of loads on debonding have been studied for the single fiber model: compression [33], residual  
65 thermal stresses [34], biaxial tension-tension and tension-compression [35], biaxial compression-compression and compression-tension [36]. The effect of the presence of nearby bonded fibers on the debonding of a fiber embedded in an infinite matrix has been studied under uniaxial transverse tension [37], biaxial

tension [38] and uniaxial transverse compression [39]. The effect of inter-fiber  
70 distance on debond growth has been studied for a partially debonded fiber at the  
center of a hexagonal cluster inside a homogenized UD composite in the case  
of fully bonded neighbouring fibers [40] and of two partially debonded fibers  
out of the surrounding six [41]. An understanding of crack shielding and finite  
thickness effects on debond growth in non-homogenized microstructural models  
75 of UD composites seems thus to be lacking: this is the problem that we want to address  
in the present work. Mode I and Mode II energy release rates will be analyzed  
using stress fields calculated with the FEM for a variety of Repeating Unit Cell  
(RUC) of the composite with square packing of fibers under transverse ten-  
sile loading. These RUCs represent composites with different distance between  
80 partially debonded fibers in the transverse direction which allows to study the  
effect of crack shielding on the ERR. In the ply thickness direction, the varying  
number of perfectly bonded fiber rows exposes the effect of the free boundary of  
the composite on debond growth. Finally, using coupling of thickness direction  
displacements on horizontal boundaries of the RUC, the accelerating effect of  
85 the interaction between debonds of fibers located on the same vertical line is  
studied.

## 2. RVE models & FE discretization

### 2.1. Introduction & Nomenclature

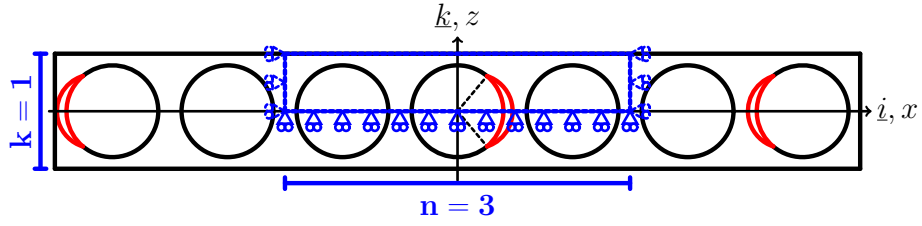
In this paper, we analyze debond development in unidirectional (UD) com-  
90 posites subjected to in-plane transverse tensile loading. The interaction between  
debonds in UD composites is studied developing models of different Repeating  
Unit Cells (RUC) of laminates where only the central fiber in the cell has a  
damage in the form of a fiber/matrix interface crack (debond). The composite  
RUC may be repeating in the in-plane transverse direction only (representing  
95 an ultra-thin composite) or repeating also in the composite thickness direction,  
representing an infinite composite in a limiting case. Thus, the conditions at  
the UD composite's upper and lower boundaries are one of the parameters for

the investigation. The used RUCs allow for considering the composite with debonds as a sequence of stacked damaged and undamaged rows, each row with only one fiber in the thickness direction. Since all of these RUCs feature regular microstructures with fibers placed according to a square-packing configuration, they are Representative Volume Elements (RVE) of composites with a certain distribution of debonds. Introducing in-plane coordinates  $x$  and  $y$ , where  $x$  is in the transverse direction of the UD composite under consideration, the strain in the  $y$ -direction due to a load in the  $x$ -direction is small, caused in turn by the very small minor Poisson's ratio of the UD composite. Additionally, debonds are considered to be significantly longer in the fiber direction than in the arc direction. Therefore, we use 2D models under the assumption of plane strain, defined in the  $x - z$  section of the composite. Thus, the analysis presented applies to long debonds, with a focus on understanding the mechanisms of growth along their arc direction. The composites are subjected to transverse tensile strain, applied as a constant displacement in the  $x$ -direction along the vertical boundary of the RUC as shown in Figure 1 to ???. As the models are differentiated by the number of rows of fibers and by the spacing between debonds along the vertical and horizontal directions, the corresponding RUCs can be distinguished from each other based on the number  $n$  of fibers in the horizontal direction and  $k$  in the vertical direction. Furthermore, the horizontal surfaces can be either free or vertical displacement coupling can be applied. We thus introduce the common notation  $n \times k - free$  and  $n \times k - coupling$  to denote a RUC with  $n \times k$  fibers and, respectively, a free upper surface or kinematic coupling applied to it. The specific combinations of particular choices of  $n$ ,  $k$ , and boundary conditions are detailed in Section 2.2, together with the description of the corresponding models of damaged composite they are representing.

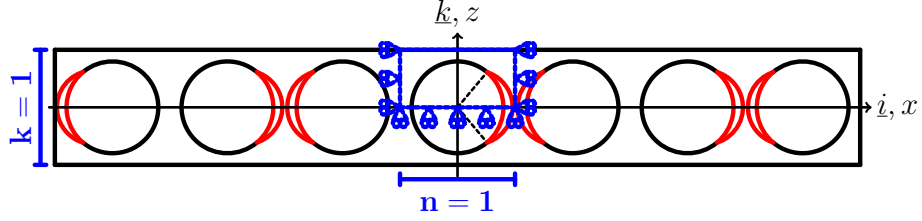
## 2.2. Models of Representative Volume Element (RVE)

The first two models feature, as shown in Fig. 1, an ultra-thin UD laminate with only one row of fibers across its thickness,  $k = 1$ . This is quite an extreme model from the microstructural point of view; however, it allows to

focus the analysis on the interaction between debonded fibers placed along the x-direction. Furthermore, as the horizontal surfaces are considered free, the interaction is stronger in this case than in any other, making the trends very clear and the predictions of this model rather conservative. In retrospective, if only 20 years ago such a model would have been considered too abstracted from the physical reality, the recent advancements in the spread tow technology make this approach appealing also as a limiting case for practical considerations.



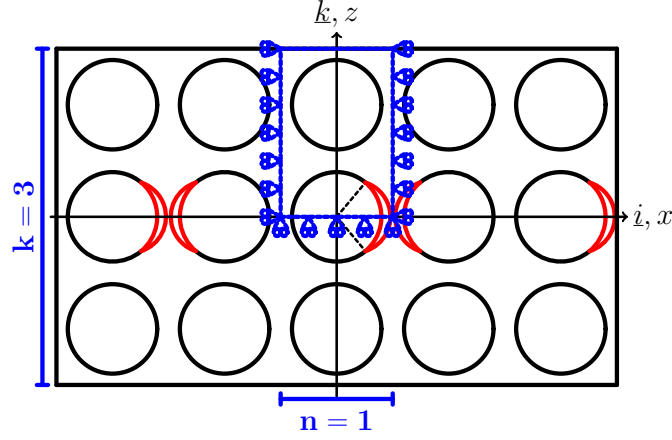
(a) Single row of fibers with a debond appearing every  $n$  fibers: model  $n \times 1 - free$  ( $n = 3$  in the figure).



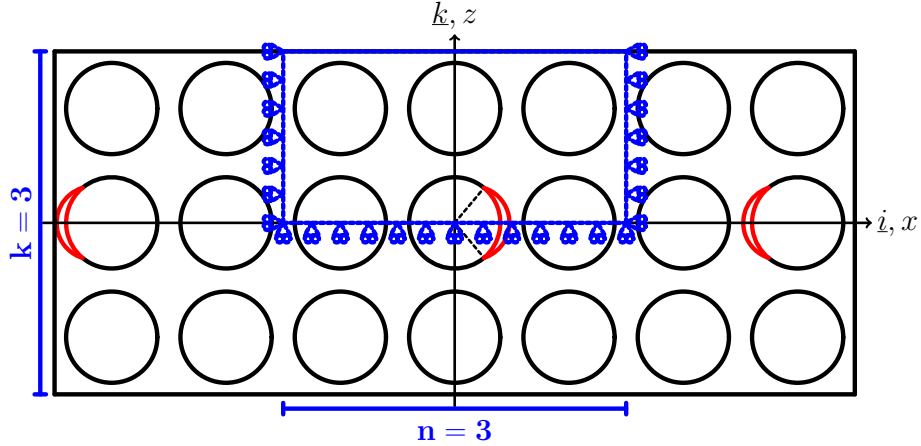
(b) Single row of fibers with debonds appearing on each fiber: model  $1 \times 1 - free$ .

Figure 1: Models of ultra-thin UD composites with a single “row” of fibers and debonds repeating at different distances. The corresponding repeating element (RUC) is highlighted in blue, while debonds are represented in red.

In the sub-model of Fig. 1a, every  $n^{th}$  fiber in the composite is partially debonded on alternating sides of the fiber. The symmetries of the model allow the use of the upper part of the RUC. It is highlighted by blue lines in Fig. 1 to 3. Following the notation introduced in Section 2.1, we will refer to this model as  $n \times 1 - free$ . In the sub-model  $n = 1$ , Fig. 1b, a debond appears on each fiber on alternating sides and the corresponding RUC contains only one fiber. We will refer to this model as  $1 \times 1 - free$ .



(a) Multiple rows of fibers with debonds appearing on each fiber belonging to the central row: model  $1 \times k - free$  ( $k = 3$  in the figure).



(b) Multiple rows of fibers with a debond appearing every  $n$  fibers within the central row: model  $n \times k - free$  ( $n = 3$  and  $k = 3$  in the figure).

Figure 2: Models of UD composites with different “rows” of fibers and debonds repeating at different distances. The corresponding repeating element (RUC) is highlighted in blue, while debonds are represented in red.

The second set of models in Fig. 2 and Fig. 3 considers laminates with multiple rows of fibers across the thickness: a finite number of rows in the first two sub-models in Fig. 2; an infinite number in the model of Fig. 3. In Fig. 2a, the RUC contains  $n = 1$  fiber in the x-direction,  $k$  fibers across the thickness

and the central fiber is debonded. This model will be referred to in the following as  $1 \times k - free$ . Thinking in terms of rows, in this model we have a central row where each fiber is debonded. This row is surrounded from each side by  $(k-1)/2$  rows with perfectly bonded fibers. In the sub-model in Fig. 2b, each  $n^{th}$  fiber in the central row is debonded and this row is surrounded by  $(k-1)/2$  rows of undamaged fibers from each side. We will refer to this model as  $n \times k - free$  (because the horizontal boundary of the RUC is free of any constraint).

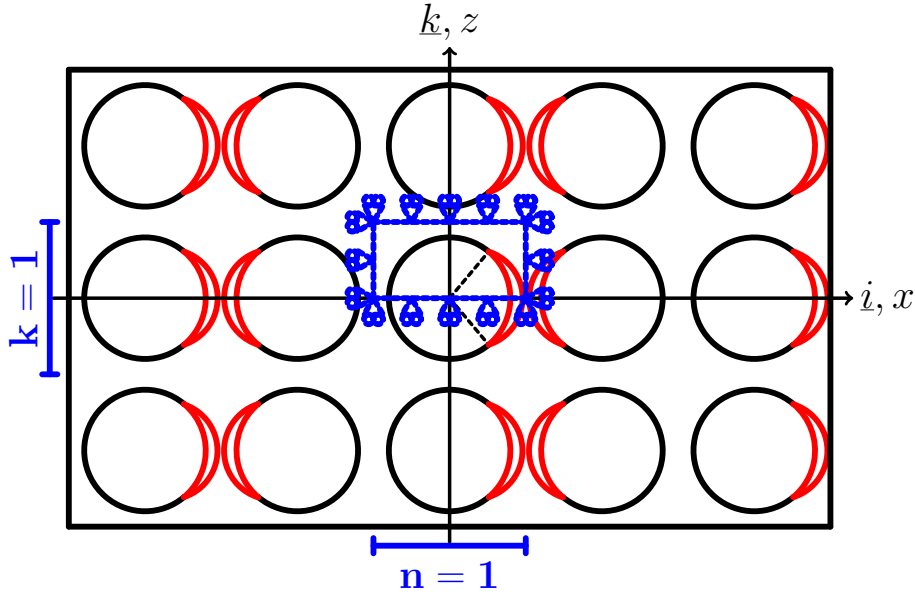


Figure 3: Model of UD composites with an infinite number of “rows” of fibers and debonds appearing on each fiber: model  $1 \times 1 - coupling$ . The corresponding repeating element (RUC) is highlighted in blue, while debonds are represented in red.

Finally, the model in Fig. 3 represents an UD composite with an infinite number of rows; all of them with partially debonded fibers. As all fibers have debonds, the corresponding RUC is made of a single partially debonded fiber with kinematic coupling conditions applied to the upper boundary to assure periodicity. This model is referred to as  $1 \times 1 - coupling$ .



### 2.3. Finite Element (FE) discretization

Each RUC is discretized using the Finite Element Method (FEM) within the Abaqus environment, a commercial FEM package [42]. The length  $l$  and height  $h$  of the model are determined by the number of fibers  $n$  in the horizontal direction and  $k$  across the thickness (see 2.2) according to Eq. 1:

$$l = 2nL \quad h = 2kL; \quad (1)$$

where  $L$  is the length of a one-fiber unit, see Fig. 4, defined as a function of the fiber volume fraction  $V_f$  and the fiber radius according to

$$L = \frac{R_f}{2} \sqrt{\frac{\pi}{V_f}}. \quad (2)$$

The fiber radius  $R_f$  is assumed to be the same for each fiber in the model and equal to  $1 \mu m$ . The latter value is not physical and it has been chosen for simplicity. It is worth to note at this point that, in a linear elastic solution as the one presented here, the ERR is proportional to the geometrical dimensions and recalculation of the ERR for fibers of any size, thus, requires a simple multiplication. Furthermore, notice that the relationships in Eqs. 1 and 2 ensure that the local and global  $V_f$  are everywhere equal.

The debond is placed symmetrically with respect to the  $x$  axis (in red in Fig. 4) and has an angular size of  $\Delta\theta$  (the full debond's size is thus  $2\Delta\theta$ ). For large debond's sizes ( $\geq 60^\circ - 80^\circ$ ), a region of variable size  $\Delta\Phi$  appears at the crack tip in which the crack faces are in contact and slide on each other. Due to its appearance, frictionless contact is considered between the two crack's faces to allow free sliding and avoid interpenetration. Symmetry with respect to the  $x$  axis is applied on the lower boundary and kinematic coupling on the  $x$ -displacement along the left and right sides. The upper boundary is in general free, except for the model  $1 \times 1$  - *coupling* (Fig. 3) which requires kinematic coupling of vertical displacements also on the upper side. Constant  $x$ -displacement  $\pm\epsilon l$ , corresponding to transverse strain  $\epsilon$  equal to 1% is applied to the right and left boundaries.

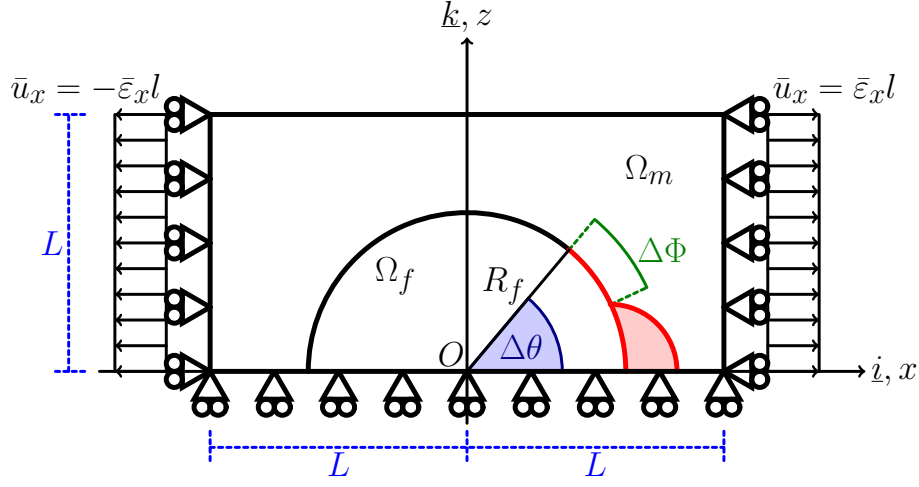


Figure 4: Schematic of the model with its main parameters.

Table 1: Summary of the mechanical properties of fiber and matrix.

Material	$E$ [GPa]	$G$ [GPa]	$\nu$ [-]
Glass fiber	70.0	29.2	0.2
Epoxy	3.5	1.25	0.4

The model is meshed using second order, 2D, plane strain triangular (CPE6) and rectangular (CPE8) elements. A regular mesh of quadrilateral elements with an almost unitary aspect ratio is required at the crack tip. The angular size  $\delta$  of an element in the crack tip region is always equal to  $0.05^\circ$ . The crack faces are modeled as element-based surfaces and a small-sliding contact pair interaction with no friction is established between them. The Mode I, Mode II and total Energy Release Rates (ERRs) (respectively referred to as  $G_I$ ,  $G_{II}$  and  $G_{TOT}$ ) represent the main output of the FEM analysis; they are evaluated using the VCCT technique [43] implemented in a custom Python routine and, for the total ERR, the J-integral [44] by application of the Abaqus built-in functionality. A glass fiber-epoxy system is considered in every model, and it is assumed that their response lies always in the linear elastic domain. The properties used are listed in Table 1.

#### 2.4. Validation of the model

The model is validated in Fig. 5 against the results reported in [45, 37], obtained with the Boundary Element Method (BEM) for a single fiber with a symmetric debond placed in an infinite matrix. This situation is modeled using the *free* RVE with  $V_f = 0.0079\%$ , which corresponds to a RUC's length and height of  $\sim 100$ .

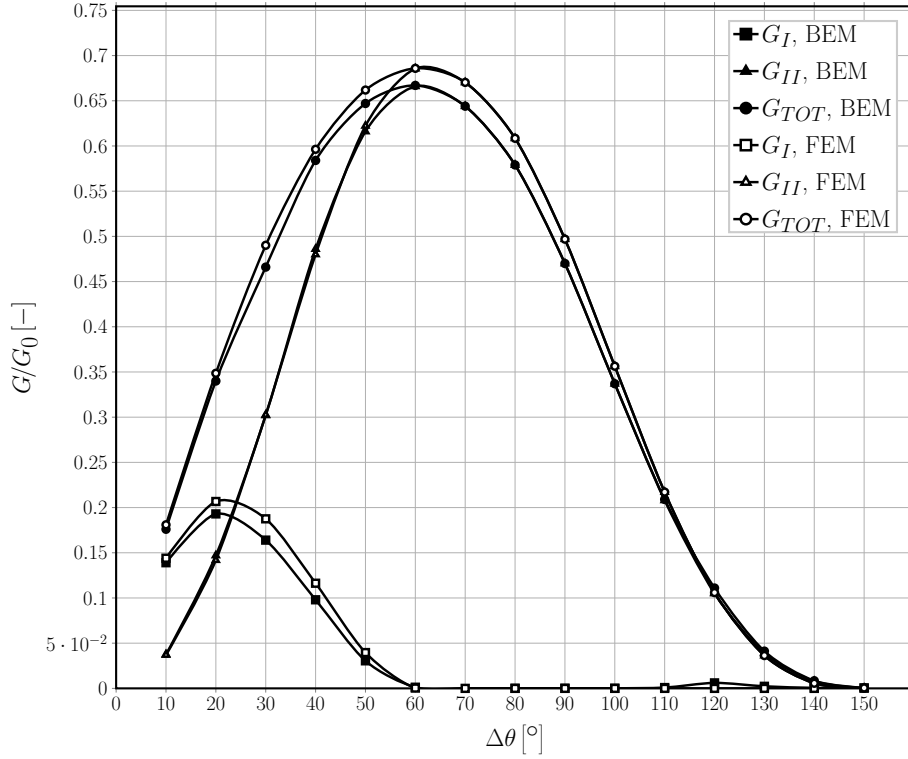


Figure 5: Validation of the single fiber model for the infinite matrix case with respect to the BEM solution in [37].

To allow for a comparison, the results are normalized following [37] with respect to a reference Energy Release Rate  $G_0$  defined as

$$G_0 = \frac{1 + k_m}{8\mu_m} \sigma_0^2 \pi R_f \quad (3)$$

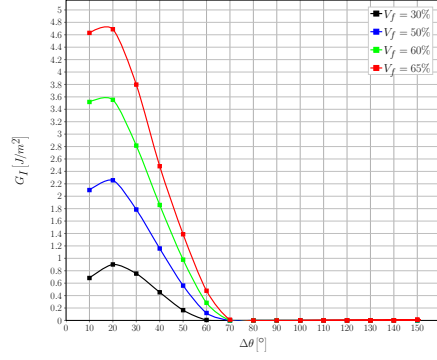
205 where  $\mu$  is the shear modulus,  $k$  is the Kolosov's constant defined as  $3 - 4\nu$   
for plane strain conditions,  $R_f$  is the fiber radius and the index  $m$  refers to  
the properties of the matrix.  $\sigma_0$  is the stress at the boundary, computed as  
the average of the stress extracted at each boundary node along the right side  
(arithmetic average as nodes are equispaced by design along both the left and  
210 right sides). The agreement is good: the difference between the BEM solution,  
which is considered more accurate, and the FEM solution does not exceed 5%.  
The ERRs' maxima are in the same positions and the size of the contact zone  
is the same. Nevertheless, an analysis of phenomena leading to less than 5%  
differences in ERR would not be reliable and, therefore, it is not recommended.

### 215 3. Results & Discussion

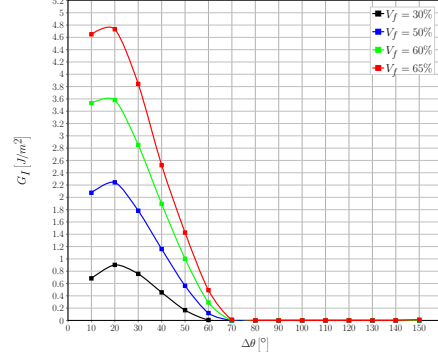
#### 3.1. Effect of Fiber Volume Fraction

As shown in Figs. 6 and 7, respectively for Mode I and Mode II, the fiber con-  
tent has a drastic effect on the Energy Release Rate at the tip of the fiber/matrix  
interface crack. The effect of four levels of fiber volume fraction are compared,  
220 30%, 50%, 60% and 65%, on two microstructural models: a  $11 \times 11$  - *free* (every  
 $11^{th}$  fiber in the central fiber row is partially debonded and, on the top of this  
row, we have 5 undamaged fiber rows), Figs. 6a and 7a, and a  $21 \times 21$  - *free*  
(every  $21^{th}$  fiber in the central fiber row is partially debonded and, on the top  
of this row, we have 10 undamaged fiber rows), Figs. 6b and 7b.

225 Comparing Fig. 6a with 6b, and Fig. 7a with 7b, we can observe that the  
ERRs' values are very similar for RUCs with  $11 \times 11$  and  $21 \times 21$  fibers, though  
they are slightly higher for the larger RUC where the next debonded fiber and  
the free surface are further away from the debonded fiber. From these results we  
conclude that both RUCs are large enough to represent a single debonded fiber  
230 in an infinite array of bonded fibers. Obviously, there exists a specific effect of  
the fiber content. For Mode I, Fig. 6, the maximum value of the ERR increases  
by  $\sim 5.2$  times when  $V_f$  changes from 30% to 65%. The debond's angular size  
for which the peak value occurs remains unchanged at  $20^\circ$ , but for  $V_f = 60\%$



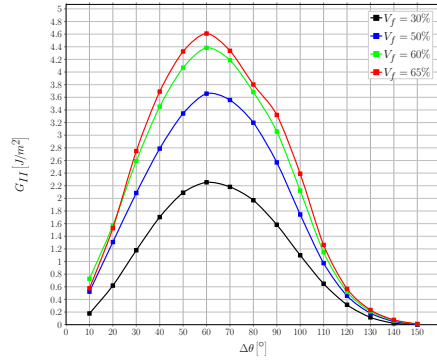
(a) Model  $11 \times 11 - free$ .



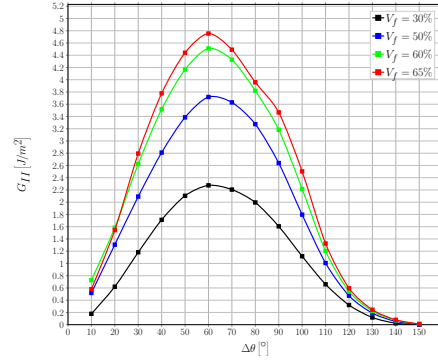
(b) Model  $21 \times 21 - free$ .

Figure 6: A view of the effect of fiber volume fraction on Mode I ERR in two exemplificative models, subject to an applied transverse strain  $\varepsilon_x$  of 1%.

and 65% the Mode I ERR at  $10^\circ$  and at  $20^\circ$  are rather similar, approximately  
 235 creating a plateau. Furthermore, increasing the fiber volume fraction delays the  
 onset of the contact zone, which corresponds in Fig. 6 to the first value of  $\Delta\theta$   
 for which  $G_I$  is equal to zero. For  $V_f = 30\%$ , the contact zone first appears for  
 a debond of  $60^\circ$ , similarly to what happens in the single fiber in infinite matrix  
 model (Fig. 5). For higher fiber contents, the contact zone's onset is delayed to  
 240 a debond's size approximately equal to  $70^\circ$ .



(a) Model  $11 \times 11 - free$ .



(b) Model  $21 \times 21 - free$ .

Figure 7: A view of the effect of fiber volume fraction on Mode II ERR in two exemplificative models, subject to an applied transverse strain  $\varepsilon_x$  of 1%.

For Mode II, Fig. 7, the maximum value of the ERR increases by  $\sim 2.1$  times when  $V_f$  changes from 30% to 65%. The effect is thus similar to Mode I, but with a significantly lower magnitude. Similar to Mode I, the debond's size for which the peak value of Mode II occurs remains unchanged, at  $60^\circ$ . There is a distinct maximum in the curve and its shape does not depend on the fiber content. It is worthwhile to notice that the ratio of Mode II to Mode I peak values is  $\frac{\max(G_{II})}{\max(G_I)} \sim \frac{2.2}{0.9} \sim 2.4$  for  $V_f = 30\%$ , while it is  $\sim \frac{4.7}{4.7} \sim 1$  for  $V_f = 65\%$ . Given that the peaks occur at different debond's sizes, for which the value of the other ERR is very small or even close to zero, this means that the increase in fiber content creates a long range of very close values of the total ERR, that may have a global destabilizing effect on the debond's growth.

The general increasing trends observed in Figs. 6 and 7 are related to the fact that, given that the global and local  $V_f$  are everywhere identical in the models presented, an increase in fiber content corresponds to a decrease in the average distance between fibers. Thus, the decay of the local stress and strain fields in the matrix domain occurs over smaller lengths causing higher values at the crack tip. The difference in relative magnitude between Mode I and Mode II and the delay in the contact zone's onset are instead due to the interplay between two different mechanisms, both caused by the ordered microstructural arrangement of the model. In the models considered, a fully bonded fiber is always placed along the horizontal direction, aligned with the partially debonded fiber and exactly in front of the debond. By increasing  $V_f$ , the former moves closer to the latter and for small debonds this causes a magnification of the  $x$ -strain at the crack tip. For small debonds ( $\leq 20^\circ - 30^\circ$ ) in fact, the crack tip is approximately normal to the  $x$ -direction and thus an increase in  $\varepsilon_x$  causes an increase in  $G_I$ . On the other hand, for large debonds ( $\geq 70^\circ - 80^\circ$ ) the crack growth direction is almost aligned with the  $x$ -axis, thus a magnification in the  $x$ -strain translates into an increase of Mode II ERR. However, this increasing effect on  $G_{II}$  is partially counteracted by the presence of a fully bonded fiber on top of the debonded fiber and aligned with it. As fibers are more rigid than the surrounding matrix, the presence of the former will restrain horizontal displace-

ments, thus hampering strong increases in  $G_{II}$  for large debonds. Furthermore, due to the mismatch in the Poisson's ratios, the fully bonded fiber placed above generates an upward-directed component of the vertical displacement field in the matrix, which tends to open the debond and causes the delay in the contact zone's onset. The interplay between these mechanisms is governed by the average inter-fiber distance and, in turn, by the fiber volume fraction.

These observations are in strong agreement with the results reported in [37], where the effect on the ERR of a partially debonded fiber of two fully bonded nearby fibers, placed symmetrically with respect to the loading direction, is studied for different angular positions (denoted as  $\theta_2$ ) and radial distances in a model with an effectively infinite matrix ( $V_f \sim 0.09\%$ ). The effect of the former is studied for a constant value of the radial distance between the debonded and bonded fibers, which corresponds to a local  $V_f^{local}$  of  $\sim 62\%$  assuming hexagonal packing. They report an increase in both Mode I and Mode II ERR with respect to the single fiber case when the two fibers are placed at an angle of respectively  $\pm 25^\circ, \pm 30^\circ, \pm 140^\circ, \pm 150^\circ, \pm 155^\circ$ , i.e. closest to the loading direction. Notice that for  $\pm 25^\circ$  and  $\pm 155^\circ$  the two fully bonded fibers are almost in contact, with an inter-fiber distance of  $\sim 0.04$  times their radius. This result confirms the considerations made in the previous paragraph about the  $x$ -strain magnification caused by the presence of fully bonded fibers along the loading direction. The effect is further analyzed and discussed in Sec. 3.2 and Sec. 3.4. In the range  $\pm 40^\circ - \pm 130^\circ$  instead, the presence of the other fibers causes a reduction of the ERR and, particularly in the range  $80^\circ - 120^\circ$ , results are very close and almost insensitive to variations in  $\theta_2$ , which supports the previous conclusion about the effect of a fully bonded fiber on top the partially debonded one. This effect is treated in more detail in Sec. 3.3.

Comparing the results from [37] with those presented in this paper, an hypothesis can be furthermore formulated about the robustness of the results of the present article with respect to deviations in fiber position: it seems reasonable to assume a tolerance to deviations of max.  $\pm 30^\circ$  with respect to the loading direction and of max.  $\pm 20^\circ$  with respect to the through-the-thickness direction.

The effect of the local fiber content is also investigated in [37], by changing the radial distance between the partially debonded fiber and the fully bonded ones. They observe that the further the fully bonded fibers are placed from the central one, i.e. the lower the local  $V_f$ , the lower is their effect on the ERR. The magnitude of the effect is however small: the maximum increase of the total ERR is of  $\sim 1.15$  times for  $\theta_2 = 30^\circ$  and  $150^\circ$  when increasing  $V_f^{local}$  from 28% to 62%; of  $\sim 1.25$  times for  $\theta_2 = 60^\circ$ ,  $\sim 1.4$  times for  $\theta_2 = 90^\circ$  and  $\sim 1.75$  times for  $\theta_2 = 120^\circ$  when decreasing  $V_f^{local}$  from 62% to 28%. Analogous results can be found in [40], where the authors consider a centrally-placed partially debonded fiber surrounded by an hexagonal cluster inside an homogenized UD composite. They observe a reduction in the ERR when the local fiber volume fraction is increased, i.e. when the spacing between fibers is reduced. The strongest change is reported for Mode II, which registers a maximum increase of  $\sim 1.36$  times when the local fiber volume fraction is decreased from 78% to 66%. Thus, the trends presented in [37, 40] agree with the results on the effect of  $V_f$  and support the considerations made so far. The stark difference in magnitude however highlights the contrast between the effect of the local  $V_f$  of a cluster of fibers inside an infinite medium and of the global  $V_f$  of long-range microstructural arrangements, such the ones considered in this article. In the former, the fiber volume fraction controls the size of a localized perturbation to a constant elastic solution; in the latter,  $V_f$  determines the characteristic lengths of a global periodic solution.

### 3.2. Interaction between debonds in UD composites with a single row of fibers

The interaction of debonds appearing at regular intervals in an ultra-thin UD composite with a single row of fibers is studied for Mode I (Fig. 8) and Mode II (Fig. 9) and fiber content equal to 30% (Figs. 8a and 9a) and 60% (Figs. 8b and 9b). The models treated are  $3 \times 1 - free$ ,  $5 \times 1 - free$ ,  $7 \times 1 - free$ ,  $11 \times 1 - free$ ,  $21 \times 1 - free$ ,  $101 \times 1 - free$  and  $201 \times 1 - free$ , corresponding respectively to a debond every  $3^{rd}$ ,  $5^{th}$ ,  $7^{th}$ ,  $11^{th}$ ,  $21^{st}$ ,  $101^{st}$  and  $201^{st}$  fiber (Fig. 1a). Given that the upper surface of the UD row is left free, the interaction



with the debonded fiber in the next RUC is stronger than in any other case and the results of this section are thus the most conservative in terms of debond's growth: the ERRs should be the largest. The effect is enhanced in composites with high  $V_f$  and especially for  $G_{II}$ : at  $V_f = 60\%$  the highest  $G_{II}$  value for the  $201 \times 1 - free$  composite in Fig. 9b is more than 3 times higher than the  $G_{II}$  value value for the  $21 \times 21 - free$  composite in Fig. 7b. Even the maximum is shifted to larger angles. The  $G_I$  value is only 30% higher.

From both Fig. 8 and Fig. 9, it can be seen that the presence of a debond close to the analyzed debond decreases the strain magnification effect discussed in Sec. 3.1 and thus reduces the value of the ERR. This phenomenon is called “crack shielding” [32].

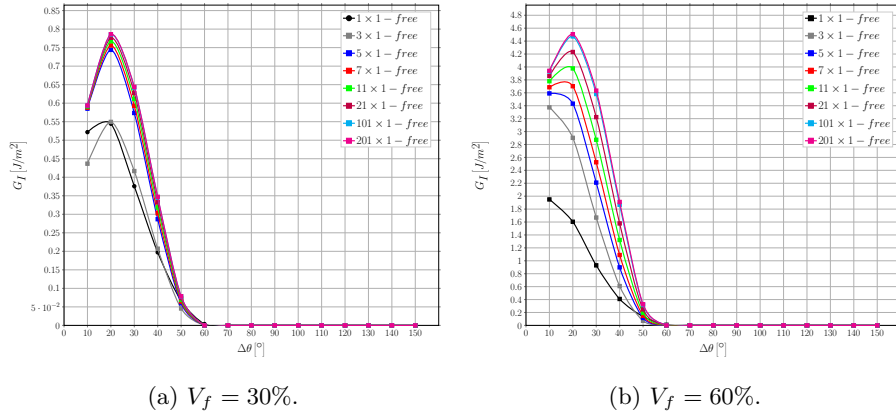


Figure 8: Effect of the interaction between debonds appearing at regular intervals on Mode I ERR in an UD with a single layer of fibers at different levels of fiber volume fraction  $V_f$ , subject to an applied transverse strain  $\varepsilon_x$  of 1%.

For Mode I, the presence of a free surface, and inversely the absence of a fully bonded fiber along the vertical direction, implies the absence of the counteracting upward-oriented vertical component of the displacement field due to the mismatch in Poisson's ratios. This in turn translates into the constancy of the value of  $\Delta\theta$  corresponding to contact zone's onset, always equal to  $60^\circ$ . For  $V_f = 30\%$ , Mode I is reduced when the spacing between debonds (in terms of number of fully bonded fibers between them in our models) decreases, but

the magnitude of change is significant only when the spacing is reduced from a debond every 5<sup>th</sup> fiber to one every 3<sup>rd</sup>. For comparison, the difference of peak  $G_{II}$  values for  $V_f = 30\%$  between  $5 \times 1 - free$  and  $3 \times 1 - free$  is  $\sim 0.2 \frac{J}{m^2}$  (around 30% of the lower value), while between  $201 \times 1 - free$  and  $5 \times 1 - free$  is  $\sim 0.05 \frac{J}{m^2}$  (around 7% of the lower value). A similar observation can be made for  $V_f = 60\%$ , but for larger spacings: no difference can be seen between the case of a debond placed every 101<sup>th</sup> and every 201<sup>th</sup> fiber. These observations suggest the existence of characteristic distance dependent on the fiber volume fraction which governs the interaction between debonds: in low  $V_f$  composites ( $V_f = 30\%$ ) the convergence to a non-interactive solution is faster (less interaction between debonded fibers in neighboring RUCs).

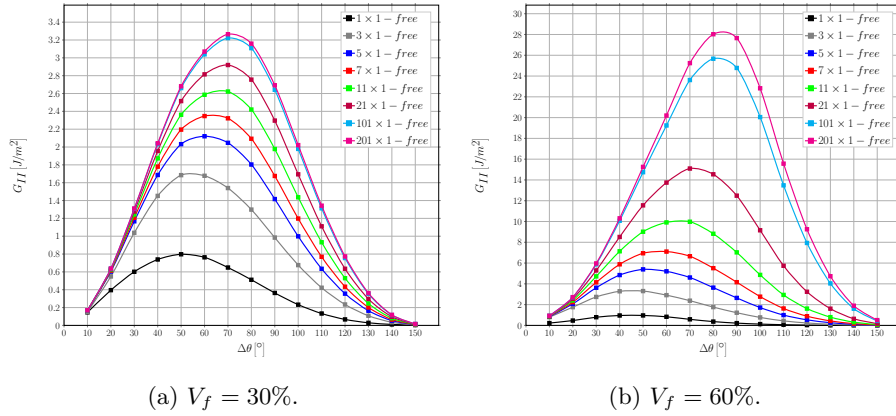


Figure 9: Effect of the interaction between debonds appearing at regular intervals on Mode II ERR in a single-ply laminate with a single layer of fibers at different levels of fiber volume fraction  $V_f$ , subject to an applied transverse strain  $\varepsilon_x$  of 1%.

Without constraint on the upper surface, the strain magnification effect creates a larger displacement gap in the  $x$ -direction, which increases Mode II for larger debonds. When debonds are far apart, the series of rigid elements in the ultra-thin composite row (constituted by fully bonded fibers and their surrounding matrix) creates higher  $x$ -strains than in average in the element with the debonded fiber, which in turn generates higher tangential displacements at the crack tip for larger debonds. Conversely, when debonds are closer (smaller num-

ber of rigid elements between them), the strain concentration in the debonded  
 370 element is more similar to the applied strain (the magnification is reduced) and  
 the tangential displacement component at the crack tip decreases for large  $\Delta\theta$ .  
 This is the mechanism behind the change in the value of  $\Delta\theta$  for which the peak  
 of  $G_{II}$  occurs: from  $70^\circ$  to  $50^\circ$  at 30%, and from  $80^\circ$  to  $40^\circ$  at 60% going  
 from the higher to the smaller spacing of debonds. Differently from Mode I,  
 375 the presence of a characteristic distance is harder to establish. For  $V_f = 30\%$   
 (Fig. 9a), it seems reasonable to establish it at around 100 fully bonded fibers  
 between each debond. For  $V_f = 60\%$  (Fig. 9b), the difference between models  
 $101 \times 1 - free$  and  $201 \times 1 - free$  is still sizable, thus preventing the estab-  
 lishment of such characteristic distance. It is possible to observe, however, that  
 380 the change between  $101 \times 1 - free$  and  $201 \times 1 - free$  is significantly smaller  
 than between  $21 \times 1 - free$  and  $101 \times 1 - free$  ( $2 \left[\frac{J}{m^2}\right]$  vs  $11 \left[\frac{J}{m^2}\right]$ ), thus sug-  
 gesting the existence of the characteristic distance outside the range studied.  
 Nevertheless, one should question whether the single row composite with free  
 surface is an appropriate RUC for defining the upper bound for  $G_{II}$ :  $G_{II}$  may  
 385 be more affected by the free surface than by the effect of the interaction between  
 debonds in the row.

### 3.3. Influence of rows of fully bonded fibers on debond's ERR in the middle row

The effect of the presence of rows of fully bonded fibers on debond's growth in  
 the central row with all fibers partially debonded is studied for Mode I (Fig. 10)  
 390 and Mode II (Fig. 11) and fiber content equal to 30% (Figs. 10a and 11a) and  
 60% (Figs. 10b and 11b). The models treated are  $1 \times 3 - free$ ,  $1 \times 5 - free$ ,  
 $1 \times 7 - free$ ,  $1 \times 11 - free$ ,  $1 \times 21 - free$ ,  $1 \times 101 - free$  and  $1 \times 201 - free$ ,  
 corresponding to a UD composite with respectively 3, 5, 7, 11, 21, 101 and 201  
 rows of fibers (Fig. 2a).

395 The results shown strengthen the arguments made in Sec. 3.1 and Sec. 3.2.  
 It can, in fact, be seen in Fig. 10 that an increasing number of bonded fiber rows  
 across the thickness delays the onset of the contact zone to a debond of  $70^\circ$  in  
 size, due to the introduction of an additional positive component of the vertical

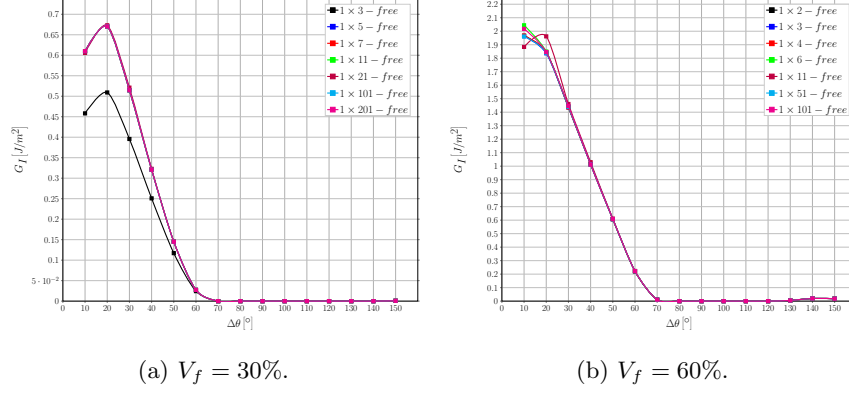


Figure 10: Influence of layers of fully bonded fibers on debond's growth in Mode I ERR in a centrally located line of debonded fibers at different levels of fiber volume fraction  $V_f$ , subject to an applied transverse strain  $\varepsilon_x$  of 1%.

displacement which translates into an opening displacement at the debond's tip. Comparing Fig. 9b with Fig. 11b, we observe that the presence of bonded fiber rows significantly reduce the  $G_{II}$  and its maximum is shifted back to  $60^\circ$ , thus confirming the hypothesis in Section 3.2 that the absence of  $G_{II}$  convergence with the increasing distance in a single-row composite is caused more by the free surface than by the interaction between debonds.

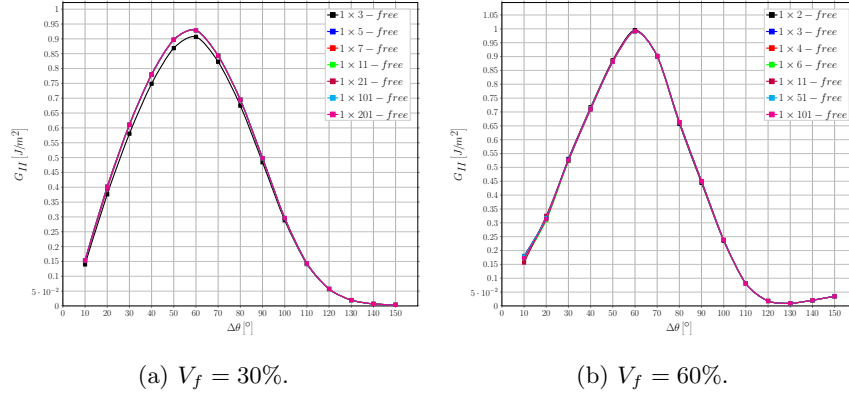


Figure 11: Influence of layers of fully bonded fibers on debond's growth in Mode II ERR in a centrally located line of debonded fibers at different levels of fiber volume fraction  $V_f$ , subject to an applied transverse strain  $\varepsilon_x$  of 1%.

405 The results of both Mode I and Mode II show that the introduction of an increasing number of fully bonded fiber rows doesn't change the ERR calculated at the crack tip (the convergence is very fast). A small effect, mostly on Mode I, of the number of bonded fiber rows can be observed at low fiber content (Figs. 10a and 11a), while for high fiber content the smaller model with only  
 410 one fiber row above the partially debonded one is already representative.

### 3.4. Interaction between debonds in the central row of a UD composite with multiple rows of bonded fibers

The interaction of debonds appearing at regular intervals in the central row of fibers in UD composites with multiple rows of fibers is investigated using  
 415 different combinations of horizontal debond spacing and the number of rows of bonded fibers across the thickness, corresponding to the models:  $3 \times 3 - free$ ,  $5 \times 3 - free$ ,  $5 \times 5 - free$ ,  $7 \times 3 - free$ ,  $7 \times 5 - free$ ,  $7 \times 7 - free$ ,  $11 \times 3 - free$ ,  $11 \times 5 - free$ ,  $11 \times 7 - free$ ,  $11 \times 11 - free$ ,  $21 \times 3 - free$ ,  $21 \times 5 - free$ ,  $21 \times 7 - free$ ,  $21 \times 11 - free$ ,  $21 \times 21 - free$ ,  $101 \times 3 - free$ ,  $101 \times 5 - free$ ,  
 420  $101 \times 7 - free$ ,  $101 \times 11 - free$ ,  $201 \times 3 - free$ ,  $201 \times 5 - free$ ,  $201 \times 7 - free$ ,  $201 \times 11 - free$  (Fig. 2b).

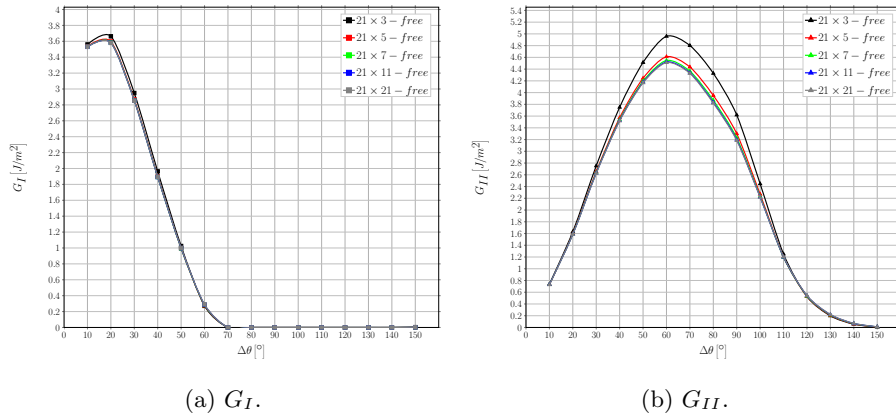


Figure 12: Effect on Mode I and Mode II ERR of the presence of an increasing number of rows of fully bonded fibers in UD composites with debonds appearing every  $10^{th}$  fiber (model  $21 \times k - free$ ).  $V_f = 60\%$  and  $\varepsilon_x = 1\%$ .

The results shown in Fig. 12 confirm the observations discussed in Sec. 3.2 and Sec. 3.3: the presence of fully bonded fiber rows on top of the central row with debonded fibers reduces the interaction with the free surface and thus has a restraining effect on the ERR, that counteracts the magnification due to an increasing number of fully bonded fibers in the horizontal direction. The interplay is further modulated by the fiber content. Observing Fig. 12, it is possible to note how the free surface interaction decays fast: the presence of 5 fiber rows across the thickness is already sufficient to prevent any significant effect on the ERR of a debond in the central row.

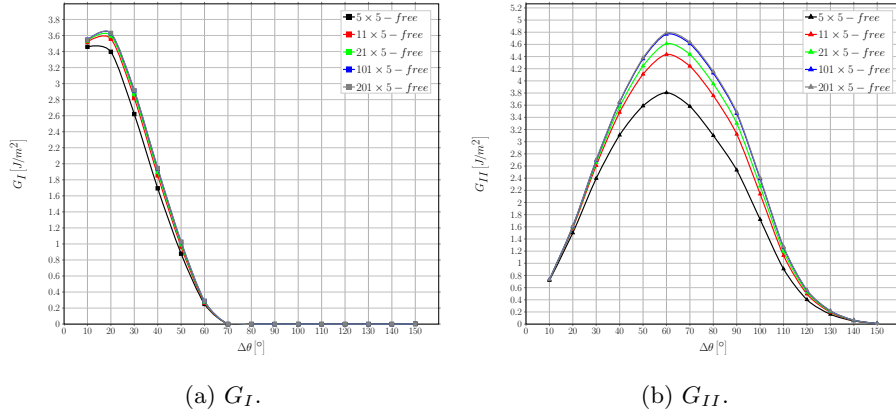


Figure 13: Effect on Mode I and Mode II ERR of increasing the spacing between debonds appearing in the central row of fibers in a UD composite with a fixed number of rows across the thickness.  $V_f = 60\%$ ,  $k = 5$  and  $\varepsilon_x = 1\%$ .

The results in Fig. 13 show instead the effect of increasing the distance between two consecutive debonds in the central row of a UD of given thickness. In agreement with the observations of Section 3.2, increasing the distance between debonds (measured in terms of fully bonded fibers) causes an increase in the ERR in both Mode I and Mode II. For both Mode I and Mode II, it is possible to observe the existence of a characteristic distance which defines the limit between the interactive and the non-interactive solution. Furthermore, comparing Figure 13a and 13b, it is possible to notice that Mode I is less sensitive than Mode II to the horizontal spacing of debonds when the free surface is far from

440 the debond itself.

### 3.5. Comparison with the single fiber model with equivalent boundary conditions

The single fiber RUC ( $1 \times 1$ –*free* or  $1 \times 1$ –*coupling*) corresponds to the most damaged state of the composite, i.e. the state in which all fibers have debonds. The  $1 \times 1$ –*free* model represents an ultra-thin UD composite with a single row of partially debonded fibers. The  $1 \times 1$ –*coupling* model, where the displacement  
445 of coupling is used to enforce periodic boundary conditions, represents an infinite composite.

The comparison of the  $1 \times 1$ –*free* model with one row multi-fiber models  $n \times 1$ –*free* in Figure 8 and Figure 9 show that the former provides in general  
450 the lowest value of the ERR (the highest crack shielding case).

The  $1 \times 1$ –*coupling* model is compared with  $1 \times 3$ –*free* and  $1 \times 201$ –*free* models in Fig. 14. In all three models the distance between debonds in the  $x$ -direction is the same and the difference is in the vertical direction. The  $1 \times 1$ –*coupling* model describes the interaction between debonds in different  
455 rows of debonded fibers whereas the  $1 \times k$ –*free* models describe the effect of the proximity of the composite’s free surface. The Mode I ERR in the  $1 \times 3$ –*free* and  $1 \times 201$ –*free* model is very similar to the  $1 \times 1$ –*coupling* model, which leads to a rather surprising conclusion. In both models we have, on the top of the central one, a large amount of fibers (bonded in one case and debonded in  
460 the other case). It appears that the effect of bonded and debonded fibers on the central debond is the same. This implies that the interaction between debonded fibers in elements placed on top of each other is small.

The same comparison for Mode II shows a sizeable difference in the range  $50^\circ - 90^\circ$ , while the results almost coincide for smaller values of  $\Delta\theta$ . The lower  
465 values of  $G_{II}$  of the  $1 \times 1$ –*coupling* model in the range  $50^\circ - 90^\circ$  are due to the shielding effect of a debond of the same size in the fiber just above the central one (modeled by the coupling boundary condition), which leaves the strip of matrix between the two fibers free to deform away from both of them due to the Poissons effect and thus favors Mode I and reduces Mode II. This translates

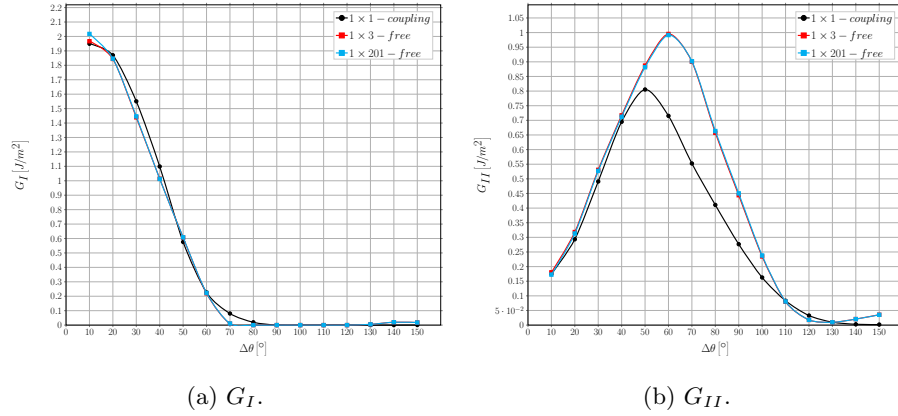


Figure 14: Comparison of the ERR between the single fiber model with coupling conditions along the upper boundary and the multiple fibers model with fibers above.  $V_f = 60\%$  and  $\varepsilon_x = 1\%$ .

into the lower estimates in Fig. 14b and into the delay in the appearance of the contact zone, particularly evident in Fig. 14a.

#### 4. Conclusions & Outlook

Several models of Repeating Unit Cell, representative of different microstructural arrangements of a unidirectional (UD) composite, have been studied in order to investigate the effect on fiber/matrix interface crack growth of the presence of partially debonded and/or fully bonded fibers. Regular microstructures based on square-packing arrangements of fibers have been loaded in transverse tension, with debonds appearing in the central row of fibers at regular intervals measured in terms of number of fully bonded fibers between them. This central row is embedded in-between a varying number of rows with perfectly bonded fibers. The surface of the composite is either traction-free or with imposed vertical displacement constraint imitating a periodic structure in the composite thickness direction.

Local and global fiber volume fractions are everywhere equal to each other by design, which establishes a direct relationship between fiber content and inter-fiber distance. The main conclusions of this work are summarized here in the



following.

1. With a decreasing number of fully bonded fibers between two partially debonded fibers in the central row, the ERR decreases. It seems to exist  
490 a characteristic distance between debonds which defines the transition to a non-interactive solution. However, this distance depends on the number of perfectly bonded fiber rows surrounding the central row.
2. The presence of a free surface close to the debond leads to higher Mode I and Mode II ERRs and a shift of the peak  $G$  values to larger debonds.
- 495 3. The presence of fibers (fully or partially bonded) in the composite thickness direction, along the same vertical line as the analyzed central fiber, appears to have a restraining effect on both  $G_I$  and  $G_{II}$ . The free composite surface effect on the ERR decays very fast: adding more than 2 fully bonded fibers below and above the central row leads to stable constant  
500 values of ERR.
4. The presence of a debond in the fiber above the central partially debonded one only delays the appearance of the contact zone, while no significant effect on the ERR has been observed.
5. Increasing the fiber content (decreasing the inter-fiber distance), magnifies  
505 in general the effects described in the previous points.
6. The results and conclusions presented agree well with previous observations reported in the literature [37, 40]. A mechanical explanation of the observed trends has been presented based on the mismatch in elastic properties, and particularly Poisson's ratios, and the positions of fibers  
510 and debonds with respect to the loading direction.

## Acknowledgements

Luca Di Stasio gratefully acknowledges the support of the European School of Materials (EUSMAT) through the DocMASE Doctoral Programme and the European Commission through the Erasmus Mundus Programme.

## 515 References

- [1] D. A. McCarville, J. C. Guzman, A. K. Dillon, J. R. Jackson, J. O. Birkland, 3.5 Design, Manufacture and Test of Cryotank Components, Elsevier, 2018, pp. 153–179. doi:10.1016/b978-0-12-803581-8.09958-6.  
URL <https://doi.org/10.1016/b978-0-12-803581-8.09958-6>
- 520 [2] Y. H. N. Kim, S. Ko, W.-S. Lay, J. Tian, P. Chang, S. U. Thielk, H.-J. Bang, J. Yang, Effects of shallow biangle, thin-ply laminates on structural performance of composite wings, AIAA Journal 55 (6) (2017) 2086–2092. doi:10.2514/1.j055465.  
URL <https://doi.org/10.2514/1.j055465>
- 525 [3] A. Kopp, S. Stappert, D. Mattsson, K. Olofsson, E. Marklund, G. Kurth, E. Mooij, E. Roorda, The aurora space launcher concept, CEAS Space Journal 10 (2) (2017) 167–187. doi:10.1007/s12567-017-0184-2.  
URL <https://doi.org/10.1007/s12567-017-0184-2>
- [4] K. Kawabe, S. Tomoda, T. Matsuo, A pneumatic process for spreading reinforcing fiber tow, in: Proceedings of the 42<sup>nd</sup> International SAMPE  
530 Symposium and Exhibition, SAMPE, pp. 65–76.
- [5] K. Kawabe, New spreading technology for carbon fiber tow and its application to composite materials, Sen'i Gakkaishi 64 (8) (2008) 262–267. doi:10.2115/fiber.64.p\_262.  
535 URL [https://doi.org/10.2115/fiber.64.p\\_262](https://doi.org/10.2115/fiber.64.p_262)
- [6] K. Kawabe, H. Sasayama, S. Tomoda, New carbon fiber tow-spread technology and applications to advanced composite materials, SAMPE Journal 45 (2) (2008) 6–17.  
URL [https://researchmap.jp/?action=cv\\_download\\_main&upload\\_id=161885](https://researchmap.jp/?action=cv_download_main&upload_id=161885)  
540 id=161885
- [7] H. Sasayama, K. Kawabe, S. Tomoda, I. Ohsawa, K. Kageyama, N. Ogata, Effect of lamina thickness on first ply failure in multidirectionally lami-

nated composites, in: Proceedings of the 8<sup>th</sup> Japan SAMPE Symposium, SAMPE, 2003.

- 545 [8] S. Tsai, S. Sihh, R. Kim, Thin ply composites, in: Proceedings of 46<sup>th</sup> AIAA/ASME/AHS/ASC Structures, Structural Dynamics & Materials Conference, 2005.
- [9] K. Yamaguchi, H. Hahn, The improved ply cracking resistance of thin-ply laminates, in: Proceedings of the 15<sup>th</sup> International Conference on  
550 Composite Materials (ICCM-15), SAMPE, 2005.
- [10] S. Sihh, R. Kim, K. Kawabe, S. Tsai, Experimental studies of thin-ply laminated composites, Composites Science and Technology 67 (6) (2007) 996–1008. doi:10.1016/j.compscitech.2006.06.008.  
URL <https://doi.org/10.1016/j.compscitech.2006.06.008>
- 555 [11] T. Yokozeiki, Y. Aoki, T. Ogasawara, Experimental characterization of strength and damage resistance properties of thin-ply carbon fiber/toughened epoxy laminates, Composite Structures 82 (3) (2008) 382–389. doi:10.1016/j.compstruct.2007.01.015.  
URL <https://doi.org/10.1016/j.compstruct.2007.01.015>
- 560 [12] T. Yokozeiki, A. Kuroda, A. Yoshimura, T. Ogasawara, T. Aoki, Damage characterization in thin-ply composite laminates under out-of-plane transverse loadings, Composite Structures 93 (1) (2010) 49–57. doi:10.1016/j.compstruct.2010.06.016.  
URL <https://doi.org/10.1016/j.compstruct.2010.06.016>
- 565 [13] J.-B. Moon, M.-G. Kim, C.-G. Kim, S. Bhowmik, Improvement of tensile properties of CFRP composites under LEO space environment by applying MWNTs and thin-ply, Composites Part A: Applied Science and Manufacturing 42 (6) (2011) 694–701. doi:10.1016/j.compositesa.2011.02.011.  
URL <https://doi.org/10.1016/j.compositesa.2011.02.011>

- 570 [14] A. Arteiro, G. Catalanotti, J. Xavier, P. Camanho, Notched response of  
non-crimp fabric thin-ply laminates, *Composites Science and Technology*  
79 (2013) 97–114. doi:10.1016/j.compscitech.2013.02.001.  
URL <https://doi.org/10.1016/j.compscitech.2013.02.001>
- [15] A. Arteiro, G. Catalanotti, J. Xavier, P. Camanho, Large damage capability  
575 of non-crimp fabric thin-ply laminates, *Composites Part A: Applied Science*  
and *Manufacturing* 63 (2014) 110–122. doi:10.1016/j.compositesa.  
2014.04.002.  
URL <https://doi.org/10.1016/j.compositesa.2014.04.002>
- [16] R. Amacher, J. Cugnoni, J. Botsis, L. Sorensen, W. Smith, C. Dransfeld,  
580 Thin ply composites: Experimental characterization and modeling of size-  
effects, *Composites Science and Technology* 101 (2014) 121–132. doi:10.  
1016/j.compscitech.2014.06.027.  
URL <https://doi.org/10.1016/j.compscitech.2014.06.027>
- [17] G. Guillaumet, A. Turon, J. Costa, J. Renart, P. Linde, J. Mayugo, Damage  
585 occurrence at edges of non-crimp-fabric thin-ply laminates under off-axis  
uniaxial loading, *Composites Science and Technology* 98 (2014) 44–50. doi:  
10.1016/j.compscitech.2014.04.014.  
URL <https://doi.org/10.1016/j.compscitech.2014.04.014>
- [18] C. Huang, S. Ju, M. He, Q. Zheng, Y. He, J. Xiao, J. Zhang, D. Jiang,  
590 Identification of failure modes of composite thin-ply laminates containing  
circular hole under tension by acoustic emission signals, *Composite Struc-*  
*tures* 206 (2018) 70–79. doi:10.1016/j.compstruct.2018.08.019.  
URL <https://doi.org/10.1016/j.compstruct.2018.08.019>
- [19] J. Cugnoni, R. Amacher, S. Kohler, J. Brunner, E. Kramer, C. Dransfeld,  
595 W. Smith, K. Scobbie, L. Sorensen, J. Botsis, Towards aerospace grade  
thin-ply composites: Effect of ply thickness, fibre, matrix and interlayer  
toughening on strength and damage tolerance, *Composites Science and*  
*Technology* 168 (2018) 467–477. doi:10.1016/j.compscitech.2018.08.

037.

600 URL <https://doi.org/10.1016/j.compscitech.2018.08.037>

- [20] H. Saito, H. Takeuchi, I. Kimpara, Experimental evaluation of the damage growth restraining in 90 layer of thin-ply cfrp cross-ply laminates, *Advanced Composite Materials* 21 (1) (2012) 57–66. doi:10.1163/156855112X629522.

- 605 [21] K. W. Garrett, J. E. Bailey, Multiple transverse fracture in 90° cross-ply laminates of a glass fibre-reinforced polyester, *Journal of Materials Science* 12 (1) (1977) 157–168. doi:10.1007/bf00738481.  
URL <https://doi.org/10.1007/bf00738481>

- [22] A. Parvizi, J. E. Bailey, On multiple transverse cracking in glass fibre epoxy cross-ply laminates, *Journal of Materials Science* 13 (10) (1978) 2131–2136.  
610 doi:10.1007/bf00541666.  
URL <https://doi.org/10.1007/bf00541666>

- [23] A. Parvizi, K. W. Garrett, J. E. Bailey, Constrained cracking in glass fibre-reinforced epoxy cross-ply laminates, *Journal of Materials Science* 13 (1)  
615 (1978) 195–201. doi:10.1007/bf00739291.  
URL <https://doi.org/10.1007/bf00739291>

- [24] J. E. Bailey, A. Parvizi, On fibre debonding effects and the mechanism of transverse-ply failure in cross-ply laminates of glass fibre/thermoset composites, *Journal of Materials Science* 16 (3) (1981) 649–659. doi:  
620 10.1007/bf02402782.  
URL <https://doi.org/10.1007/bf02402782>

- [25] J. E. Bailey, P. T. Curtis, A. Parvizi, On the transverse cracking and longitudinal splitting behaviour of glass and carbon fibre reinforced epoxy cross ply laminates and the effect of poisson and thermally generated strain, *Proceedings of the Royal Society A: Mathematical, Physical and Engineering*  
625

Sciences 366 (1727) (1979) 599–623. doi:10.1098/rspa.1979.0071.

URL <https://doi.org/10.1098/rspa.1979.0071>

- [26] A. H. England, An arc crack around a circular elastic inclusion, Journal of Applied Mechanics 33 (3) (1966) 637. doi:10.1115/1.3625132.

630 URL <https://doi.org/10.1115/1.3625132>

- [27] A. Perlman, G. Sih, Elastostatic problems of curvilinear cracks in bonded dissimilar materials, International Journal of Engineering Science 5 (11) (1967) 845–867. doi:10.1016/0020-7225(67)90009-2.

URL [https://doi.org/10.1016/0020-7225\(67\)90009-2](https://doi.org/10.1016/0020-7225(67)90009-2)

- 635 [28] M. Toya, A crack along the interface of a circular inclusion embedded in an infinite solid, Journal of the Mechanics and Physics of Solids 22 (5) (1974) 325–348. doi:10.1016/0022-5096(74)90002-7.

URL [https://doi.org/10.1016/0022-5096\(74\)90002-7](https://doi.org/10.1016/0022-5096(74)90002-7)

- [29] M. Comninou, The interface crack, Journal of Applied Mechanics 44 (4) (1977) 631. doi:10.1115/1.3424148.

640 URL <https://doi.org/10.1115/1.3424148>

- [30] F. París, J. C. Caño, J. Varna, The fiber-matrix interface crack — a numerical analysis using boundary elements, International Journal of Fracture 82 (1) (1996) 11–29. doi:10.1007/bf00017861.

645 URL <https://doi.org/10.1007/bf00017861>

- [31] J. Varna, F. París, J. C. Caño, The effect of crack-face contact on fiber/matrix debonding in transverse tensile loading, Composites Science and Technology 57 (5) (1997) 523–532. doi:10.1016/s0266-3538(96)00175-3.

650 URL [https://doi.org/10.1016/s0266-3538\(96\)00175-3](https://doi.org/10.1016/s0266-3538(96)00175-3)

- [32] I. García, V. Mantić, E. Graciani, Debonding at the fibre–matrix interface under remote transverse tension. one debond or two symmetric debonds?, European Journal of Mechanics - A/Solids 53 (2015) 75–88. doi:10.1016/

j.euomechsol.2015.02.007.

655 URL <https://doi.org/10.1016/j.euomechsol.2015.02.007>

- [33] E. Correa, E. Gamstedt, F. París, V. Mantič, Effects of the presence of compression in transverse cyclic loading on fibre–matrix debonding in uni-directional composite plies, *Composites Part A: Applied Science and Manufacturing* 38 (11) (2007) 2260–2269. doi:10.1016/j.compositesa.2006.11.002.

660 URL <https://doi.org/10.1016/j.compositesa.2006.11.002>

- [34] E. Correa, V. Mantič, F. París, Effect of thermal residual stresses on matrix failure under transverse tension at micromechanical level: A numerical and experimental analysis, *Composites Science and Technology* 71 (5) (2011) 622–629. doi:10.1016/j.compscitech.2010.12.027.

665 URL <https://doi.org/10.1016/j.compscitech.2010.12.027>

- [35] E. Correa, F. París, V. Mantič, Effect of the presence of a secondary transverse load on the inter-fibre failure under tension, *Engineering Fracture Mechanics* 103 (2013) 174–189. doi:10.1016/j.engfracmech.2013.02.026.

670 URL <https://doi.org/10.1016/j.engfracmech.2013.02.026>

- [36] E. Correa, F. París, V. Mantič, Effect of a secondary transverse load on the inter-fibre failure under compression, *Composites Part B: Engineering* 65 (2014) 57–68. doi:10.1016/j.compositesb.2014.01.005.

URL <https://doi.org/10.1016/j.compositesb.2014.01.005>

- 675 [37] C. Sandino, E. Correa, F. París, Numerical analysis of the influence of a nearby fibre on the interface crack growth in composites under transverse tensile load, *Engineering Fracture Mechanics* 168 (2016) 58–75. doi:10.1016/j.engfracmech.2016.01.022.

URL <https://doi.org/10.1016/j.engfracmech.2016.01.022>

- 680 [38] C. Sandino, E. Correa, F. París, Composite materials under transverse biaxial loads: Study of the influence of a nearby fibre on the interface crack

growth under tension, in: Proceeding of the 17<sup>th</sup> European Conference on Composite Materials (ECCM-17), 2016.

[39] C. Sandino, E. Correa, F. París, Interface crack growth under transverse compression: nearby fibre effect, in: Proceeding of the 18<sup>th</sup> European Conference on Composite Materials (ECCM-18), 2018.

[40] L. Zhuang, A. Pupurs, J. Varna, R. Talreja, Z. Ayadi, Effects of inter-fiber spacing on fiber-matrix debond crack growth in unidirectional composites under transverse loading, Composites Part A: Applied Science and Manufacturing 109 (2018) 463–471. doi:10.1016/j.compositesa.2018.03.031.

URL <https://doi.org/10.1016/j.compositesa.2018.03.031>

[41] J. Varna, L. Q. Zhuang, A. Pupurs, Z. Ayadi, Growth and interaction of debonds in local clusters of fibers in unidirectional composites during transverse loading, Key Engineering Materials 754 (2017) 63–66. doi:10.4028/www.scientific.net/kem.754.63.

URL <https://doi.org/10.4028/www.scientific.net/kem.754.63>

[42] Simulia, Providence, RI, USA, ABAQUS/Standard User’s Manual, Version 6.12 (2012).

[43] R. Krueger, Virtual crack closure technique: History, approach, and applications, Applied Mechanics Reviews 57 (2) (2004) 109. doi:10.1115/1.1595677.

URL <https://doi.org/10.1115/1.1595677>

[44] J. R. Rice, A path independent integral and the approximate analysis of strain concentration by notches and cracks, Journal of Applied Mechanics 35 (2) (1968) 379. doi:10.1115/1.3601206.

URL <https://doi.org/10.1115/1.3601206>

[45] F. París, E. Correa, V. Mantić, Kinking of transversal interface cracks between fiber and matrix, Journal of Applied Mechanics 74 (4) (2007) 703.



doi:10.1115/1.2711220.

URL <https://doi.org/10.1115/1.2711220>