

ESTIMATING THE AVERAGE SIZE OF FIBER/MATRIX INTERFACE CRACKS IN UD AND CROSS-PLY LAMINATES

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Summary: *Initiation and propagation of fiber/matrix interface cracks are analyzed in Representative Volume Elements (RVEs) of UD and cross-ply laminates. By studying the distribution of stresses at the fiber/matrix interface in the undamaged case, an estimate of the initial flaw size is derived. By adopting a 2-parameters energy-based criterion for propagation [1], we then proceed to the estimation of the expected average debond size in different microstructural arrangements. Finally, the results are compared with microscopic observations available in the literature [2].*

1. INTRODUCTION

Characterization of initiation and growth of fiber/matrix interface cracks (or debonds) in Fiber Reinforced Polymer (FRP) composites has proven elusive, particularly the determination of initial flaw size and maximum debond extension. Different methodological approaches exist in the literature to study debond growth, namely Linear Elastic Fracture Mechanics (LEFM) [3], Finite Fracture Mechanics (FFM) [4], Cohesive Zone Modeling (CSM) [5], as well as several initiation physics-based criteria, such as tensile failure of the interface [3], local cavitation-induced brittle matrix failure [6, 7], local matrix plasticization [5], local nucleation plane [8]. We propose here an attempt to evaluate initiation and propagation of debonds by taking into account the uncertainty in failure criteria and mechanisms evident from the literature. We thus aim to estimate a range of plausible values of initial flaw size and maximum debond size by: first, analyzing the stress distribution at the fiber/matrix interface in the absence of damage in Representative Volume Elements (RVEs) of UD and cross-ply laminates; second, determining the angular size of the regions where $\sigma \geq 99\%\sigma_{max}$, $\sigma \geq 95\%\sigma_{max}$ and $\sigma \geq 90\%\sigma_{max}$ which σ_{max} the stress peak value, which provides an estimate of the expected initial flaw size; third, estimating the Mode I critical Energy Release Rate (ERR) of the interface crack, based on the values of initial flaw size and assuming a mixed-mode energy-based debond propagation criterion [1]; fourth, comparing total ERR with the critical ERR provided by the propagation criterion [1]. Finally, the estimated range of maximum debond size is compared with the results of microscopical observations of cross-ply specimens reported in [2].

2. RVE MODELS AND FE DISCRETIZATION

In this contribution, we analyze debond initiation and propagation in Representative Volume Elements (RVEs) of Uni-Directional (UD) composites and $[0_{m \cdot k \cdot 2L}^\circ, 90_{k \cdot 2L}^\circ, 0_{m \cdot k \cdot 2L}^\circ]$ laminates. Given a global reference frame with axis x , y and z , both types of composites are modeled as plates lying in the $x - y$ plane, with the through-the-thickness direction thus aligned with the z axis. The UD composite 0° direction is parallel to the y axis, while the cross-ply 0° direction is parallel to the x axis. Both composites are loaded in tension along the x axis, which thus corresponds to: transverse loading of the UD specimen; axial loading of the cross-ply specimen. In both composites, damage is present only in the form of fiber/matrix interface cracks, or debonds. In cross-ply, debonds are assumed to be present only in the central 90° ply. Given that: first, in the presence of a load in the x -direction, in both lay-ups the y -strain due to Poisson's effect is very small; second, debond size is assumed to be considerably larger in the fiber than in the arc direction [9]; we can consider 2D models under plane strain conditions defined in the $x - z$ plane. Although generalized plane strain conditions would represent a more appropriate modeling choice, it would limit the ability to compare with previous results. Simple plane strain conditions are thus preferred. UD composites and 90° plies are characterized by a regular microstructure following a square-packing configuration of fibers, built through the repetition of a one-fiber unit cell along the horizontal and the vertical direction. This unit cell is a square with the center occupied by one fiber of radius $R_f = 1 \mu m$ and the rest of the element constituted by matrix. The size of the one-fiber unit cell is $2L \times 2L$, such that:

$$L = \frac{R_f}{2} \sqrt{\frac{\pi}{V_f}}, \quad (1)$$

where V_f is the fiber volume fraction, here assumed to be 60%. It is worth to specify at this point that the choice $R_f = 1 \mu m$ is arbitrary and stems from the fact that ERR, in the contest of a linear elastic solution as the one considered in this article, is proportional to the geometrical dimensions of the model. Simplicity is thus the main reason for this choice. Also, V_f is always the same in the one-fiber unit cell and the entire RVE, i.e. no fiber clustering is analyzed in this work. In the case of cross-ply laminates, the 0° layer is homogenized with properties evaluated according to the Concentric Cylinders Assembly with Self-Consistent Shear (CCA-SCS) model [10, 11]. A glass fiber-epoxy system is considered for both UD and cross-ply laminates. Material properties are reported in Table 1.

Table 1. Summary of mechanical properties of fiber, matrix and UD layer.

Material	V_f [%]	E_L [GPa]	E_T [GPa]	G_{LT} [GPa]	ν_{LT} [—]	ν_{TT} [—]
Glass fiber	-	70.0	70.0	29.2	0.2	0.2
Epoxy	-	3.5	3.5	1.25	0.4	0.4
UD	60.0	43.442	13.714	4.315	0.273	0.465

The use of coupling conditions allows the study of a Repeating Unit Cells (RUC) of reduced size with respect to the corresponding RVE, which translates in a gain in terms of computational time and memory usage during the evaluation of Finite Element (FE) solution. The RVEs studied in this article are reported in Figure 1 with the corresponding RUC highlighted by dashed line (in blue in the online color version) and with symmetry and coupling conditions represented by rollers (♻️). Details about the central one-fiber unit cell are shown in Figure 2. Notice that the analysis, in terms of stresses and Energy Release Rate (ERR), is conducted on this central one-fiber unit cell, both in the case of an undamaged and of a partially debonded fiber.

Nomenclature and main features of the RVEs studied are described in the following.

$n \times k$ – free, Figure 1(a): UD composite with thickness $t_{UD} = k \cdot 2L$, where k is the number of fiber “rows” in the vertical (through-the-thickness) direction and $2L$ the side length of the one-fiber unit cell as defined in Equation 1. Debonds appear only in the central fiber “row” with $n - 1$ fully bonded fibers in between, where n is the number of fibers present in the RUC along the horizontal direction.

$n \times k$ – coupling, Figure 1(b): UD composite with infinite thickness. Debonds appear every $k - 1$ fully bonded fibers along the vertical direction in a fiber “column”, fiber “columns” containing debonds appear every $n - 1$ “columns” of fully bonded fibers. Debonds are placed on the same side of the partially debonded fiber inside the same fiber “column”, while they appear on alternating sides between one fiber “column” with damage to the next. The RUC has thus n fibers in the horizontal and k fibers in the vertical direction. Conditions of coupling of the vertical displacement are applied on the top side.

$n \times k$ – asymm, Figure 1(c): UD composite with infinite thickness. Debonds appear every $k - 1$ fully bonded fibers along the vertical direction in a fiber “column”, fiber “columns” containing debonds appear every $n - 1$ “columns” of fully bonded fibers. Debonds are placed on opposite sides of the partially debonded fiber inside the same fiber “column” and they appear on alternating sides between one fiber “column” with damage to the next. The RUC has thus n fibers in the horizontal and k fibers in the vertical direction. The following set of conditions is applied to the upper boundary:

$$\begin{aligned} u_z(x, kL) - u_z(0, kL) &= -(u_z(-x, kL) - u_z(0, kL)) \\ u_x(x, kL) &= -u_x(-x, kL) \end{aligned} \quad (2)$$

which represent conditions of anti-symmetric coupling [12].

$n \times k$ – $m \times t_{90^\circ}$, Figure 1(d): Cross-ply laminate with 90° layer thickness $t_{90^\circ} = k \cdot 2L$ and 0° layer thickness $t_{0^\circ} = m \cdot t_{90^\circ}$. k is the number of fiber “rows” in the vertical (through-the-thickness) direction of the 90° layer and $2L$ the side length of the one-fiber unit cell as defined in Equation 1. Debonds are present only in the central fiber “row” of

the 90° layer with $n - 1$ fully bonded fibers in between, where n is the number of fibers present in the RUC along the horizontal direction.

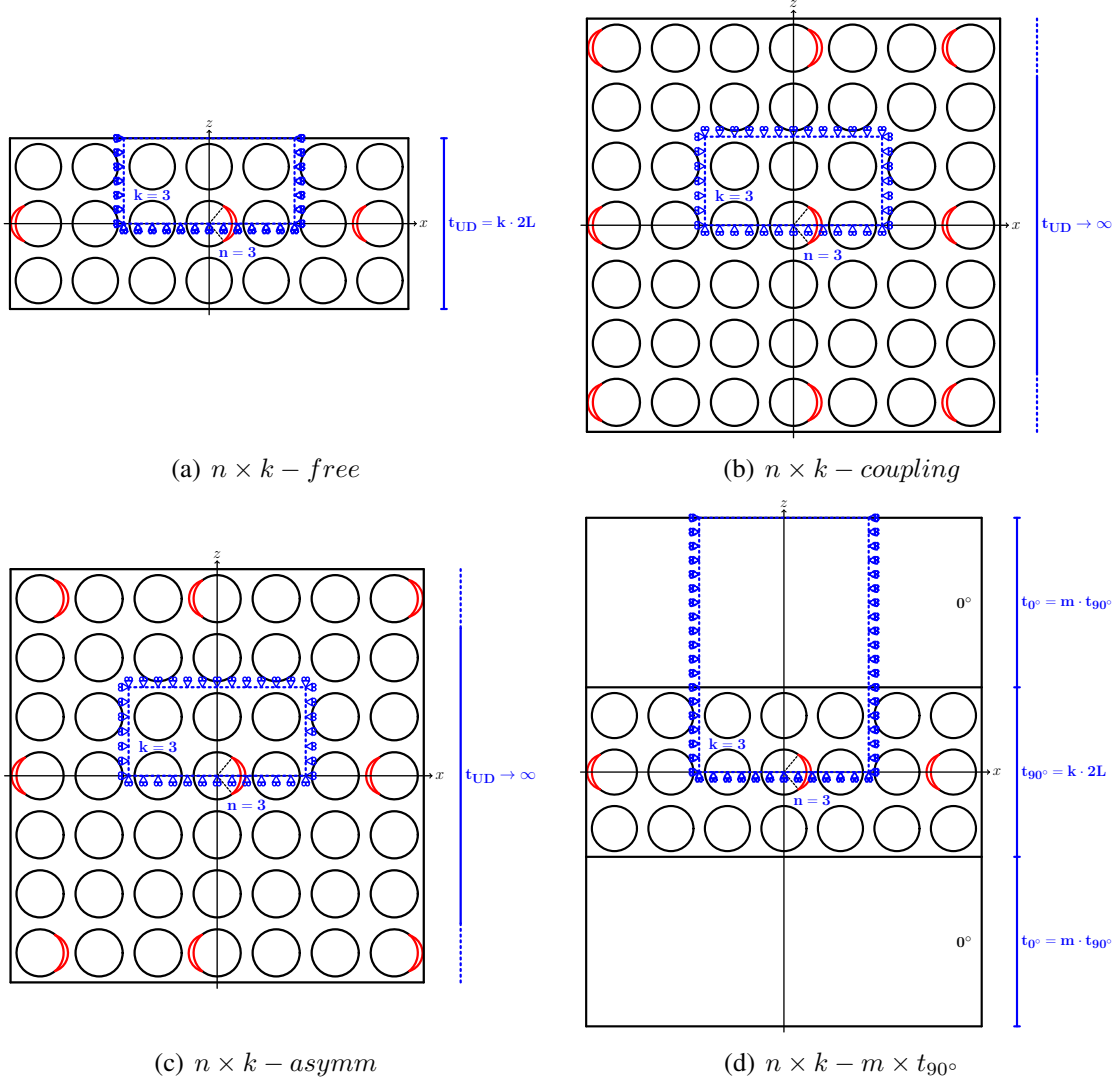


Figure 1. Composite RVEs and corresponding RUCs analyzed.

Every RUC is symmetric with respect to the horizontal (x) direction, thus only half of it is modeled in the FE solution through the use of symmetry boundary conditions on the bottom side. Conditions of coupling of the horizontal displacement are applied on the left and right side, to model the repetition of the RUC along the horizontal direction. A tensile load is applied on the right and left side in the form of displacement $\bar{u}_x = \pm \bar{\epsilon}_x nL$ with $\bar{\epsilon}_x = 1\%$. The debond has a size of $2\Delta\theta$ (see Figure 2), with $\Delta\theta \geq 0$ ($\Delta\theta = 0$ is the case of no damage at all). For large debonds ($\Delta\theta \geq 60^\circ - 80^\circ$), a region called *contact zone*, of size $\Delta\Phi$ to be determined

by the solution itself, appears at the crack tip. Correct resolution of this behavior requires the imposition of conditions of non-interpenetration of the crack faces. Crack face contact is assumed to be frictionless.

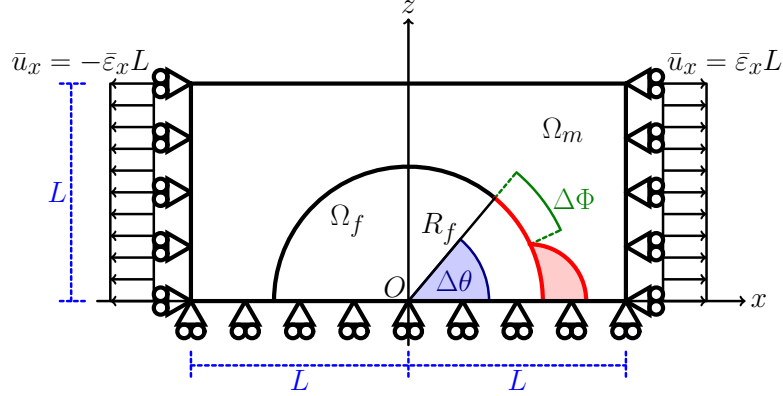


Figure 2. One-fiber unit cell and main parameters characterizing the debonding process.

The FE solution is obtained using Abaqus [13] using second order, 2D, plane strain triangular (CPE6) and rectangular (CPE8) elements. To accurately resolve the singularity at the crack tip, a regular mesh of only rectangular elements is used with almost unitary aspect ratio and angular size $\delta = 0.05^\circ$. The crack faces are represented as element-based surfaces with frictionless small-sliding contact pair interaction.

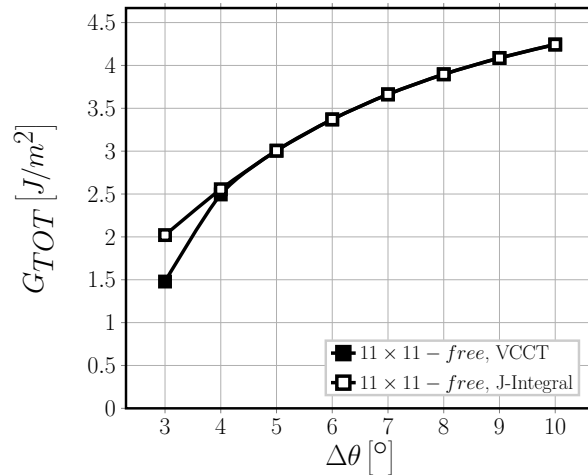


Figure 3. Comparison of total ERR of $11 \times 11 - free$ evaluated respectively with VCCT and J-integral.

Mode I ERR is underestimated with the Virtual Crack Closure Technique (VCCT) [14] for very small debonds (see Figure 3), due to the high $\frac{\delta}{\Delta\theta}$ ratio. Thus, for $\Delta\theta < 10^\circ$, G_{TOT} is evaluated using the J-integral [15], G_{II} with the VCCT and $G_I = G_{TOT} - G_{II}$; for $\Delta\theta \geq 10^\circ$ only the VCCT is used. Validation of the model is performed with respect to BEM results

of [3, 16]; the order of accuracy of the results (differences $\leq 5\%$ are not significant) is discussed in [17].

3. STRESS-BASED ANALYSIS OF DEBOND INITIATION

The distribution of stresses at the fiber/matrix interface is analyzed in models $1 \times k - free$ and $1 \times k - 1 \cdot t_{90^\circ}$ with $k = 1, 3, 11, 201$ and $\Delta\theta = 0^\circ$, i.e. the undamaged case. Some selected stress components are analyzed, based on their relevance in previous studies on debond initiation and growth: the radial σ_{rr} and shear $\tau_{r\psi}$ stress [4] (Fig. 4(a) and Fig. 4(b)), the Local Hydrostatic Stress (LHS, following the notation of [18]) σ_{LHS} [6, 7] (Fig. 4(c) and Fig. 4(d)), the local von Mises stress σ_{vM} [5] (Fig. 4(e) and Fig. 4(f)), the Local Maximum Principal Stress (LMPS, following the notation of [18]) σ_{LMPS} [8] (Fig. 4(g) and Fig. 4(h)). In plane strain conditions there exists an out-of-plane axial component of the stress (σ_{yy} in our notation): in order to study the importance of this out-of-plane component (tri-axial stress state, see [19]), σ_{LHS} , σ_{vM} and σ_{LMPS} are evaluated both neglecting (index $2D$) and considering (index $3D$) σ_{yy} .

It is possible to observe that:

1. for all stress components, no significant difference is present between the different RUCs for $\psi \leq 10^\circ$;
2. for all stress components, no difference can be observed by increasing k when $k \geq 3$;
3. for all stress components, no difference can be observed between $1 \times k - free$ and $1 \times k - 1 \cdot t_{90^\circ}$ for $k \geq 3$;
4. at $\psi = 0^\circ$ (i.e. the intersection between fiber/matrix interface and horizontal axis); the shear stress $\tau_{r\psi}$ (Fig. 4(b)) is 0 while the radial stress σ_{rr} (Fig. 4(a)) is at its maximum;
5. comparison of Fig. 4(c) with Fig. 4(d) shows that the out-of-plane stress σ_{yy} has only a marginal effect on the local hydrostatic stress σ_{LHS} : the distribution remains the same, while the peak value is reduced from 270 [MPa] in Fig. 4(c) to 250 [MPa] in Fig. 4(d) by considering σ_{yy} ;
6. comparison of Fig. 4(e) with Fig. 4(f) shows a remarkable effect of the out-of-plane stress σ_{yy} on the von Mises stress: if σ_{yy} is neglected, σ_{vM} presents a maximum of 287 [MPa] at $\psi = 0^\circ$, slightly higher than the value of σ_{LHS} (neglecting σ_{yy}) of 270 [MPa] at $\psi = 0^\circ$; if σ_{yy} is considered, the peak value of σ_{vM} is shifted to $\psi \sim 20^\circ$ and reduced to 186 [MPa] while at $\psi = 0^\circ$ σ_{vM} is reduced to ~ 110 [MPa], significantly lower than $\sigma_{LHS} = 250$ [MPa] at $\psi = 0^\circ$ (considering σ_{yy});
7. comparison of Fig. 4(g) with Fig. 4(h) shows that the Local Maximum Principal Stress is practically unaffected by the out-of-plane stress σ_{yy} ;
8. σ_{rr} , $\sigma_{LHS,2D}$, $\sigma_{LHS,3D}$, $\sigma_{vM,2D}$, $\sigma_{LMPS,2D}$ and $\sigma_{LMPS,3D}$ all reach their peak value at 0° and 180° and decrease to 99% the peak value between 2° and 8° , to 95% the peak value between 6° and 12° and to 90% the peak value between 8° and 15° from the occurrence of the maximum.

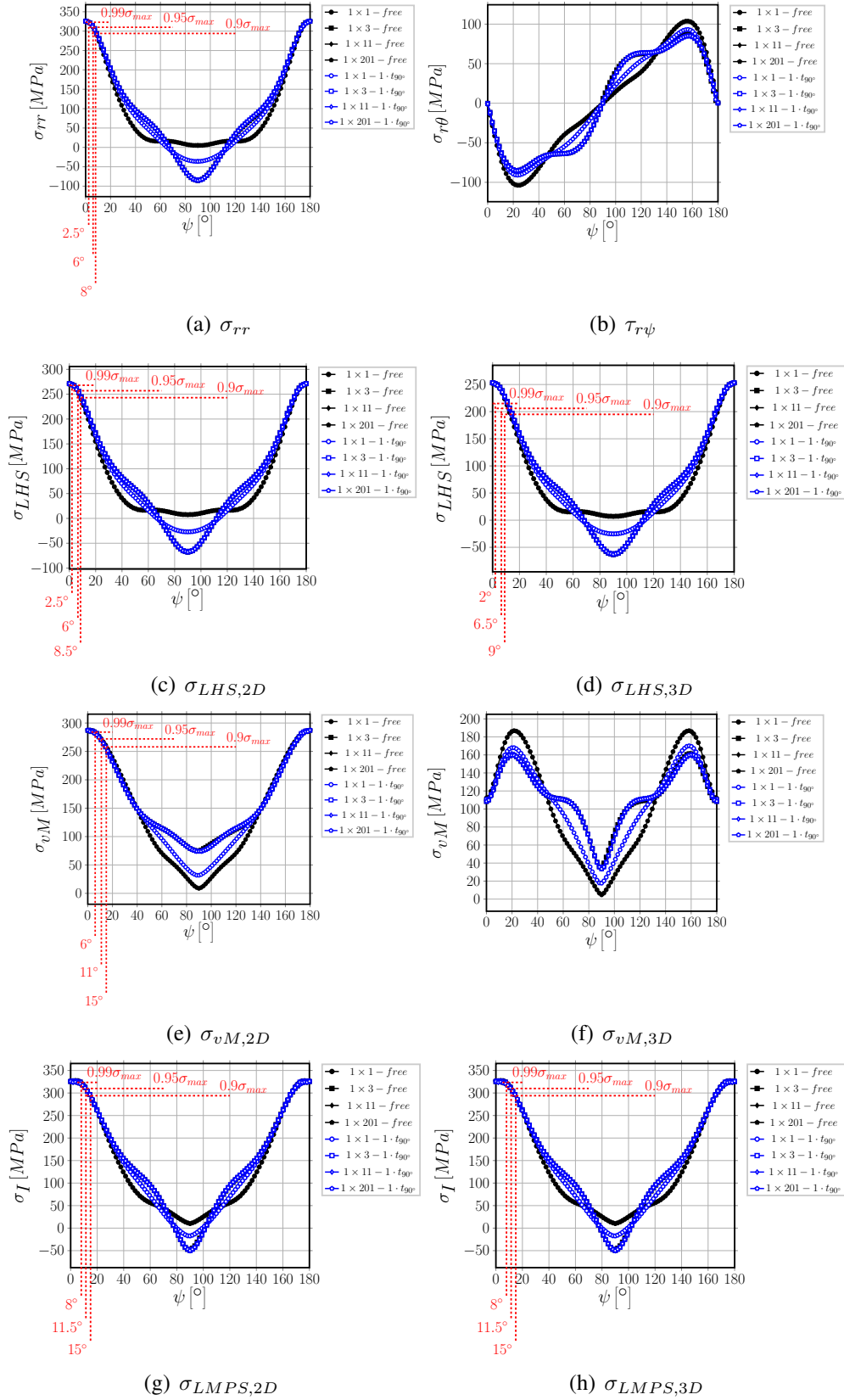


Figure 4. Stress distribution at the fiber/matrix interface in the absence of damage.

Based on the previous considerations, it is reasonable to assume that a stress-based criterion would predict, irrespectively of the specific criterion chosen, the onset of an interface crack at 0° or 180° with an initial size at least comprised in the range $2^\circ - 8^\circ$ (1% margin) and likely in the range $6^\circ - 12^\circ$ (5% margin).

4. ENERGY-BASED ANALYSIS OF DEBOND PROPAGATION

We assume an expression for the critical Energy Release Rate G_c at the interface of the form [1, 4]:

$$G_c = G_{Ic} \left(1 + \tan^2 \left((1 - \lambda) \Psi_G \right) \right), \quad \Psi_G = \tan^{-1} \left(\sqrt{\frac{G_{II}}{G_I}} \right), \quad (3)$$

where Ψ_G is the energy-based phase angle, G_{Ic} is the Mode I critical ERR (material property) and λ the mode mixity sensitivity parameter (material property, usually $0.2 \leq \lambda \leq 0.35$)¹. By calculating Mode I, Mode II and total ERR and equating the expression of G_c in Eq. 3 to G_{TOT} , it is possible to estimate the value of G_{Ic} for different values of the assumed initial debond size $\Delta\theta_0$ and of the parameter λ (Figure 5). The estimation is performed using a restricted set of RUCs: $11 \times 1 - free$, $11 \times 11 - free$, $11 \times 1 - 1 \cdot t_{90^\circ}$ and $11 \times 11 - 1 \cdot t_{90^\circ}$.

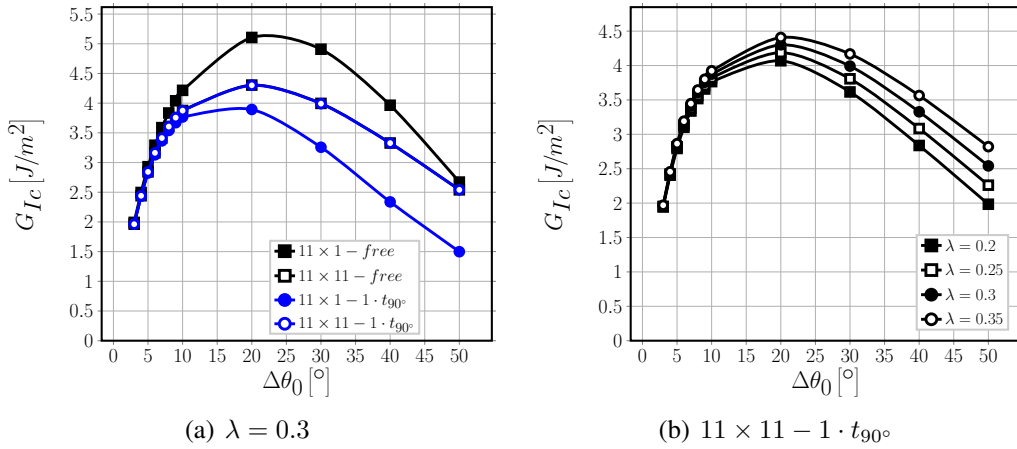


Figure 5. Estimated values of G_{Ic} assuming different values of initial debond size, for different RUCs and varying λ .

Remarkably, neither RUC configuration nor λ affects G_{Ic} significantly for small initial flaw size ($\Delta\theta_0 < 10^\circ$). Based on the considerations of the previous section, we can expect the initial flaw to be in the range $2^\circ - 12^\circ$ (1 – 5% margin): comparing this with the results of Figure 5, we can assume G_{Ic} to be in the range $2 - 4.5 \left[\frac{J}{m^2} \right]$, while $0.2 \leq \lambda \leq 0.35$.

¹ λ governs the influence of mode ratio $\frac{G_{II}}{G_I}$ on G_c such that: for $\lambda = 1$, $G_c = G_{Ic}$; for $\lambda = 0$, $G_c = G_{Ic} \left(1 + \frac{G_{II}}{G_I} \right)$.

Having estimated the values of the material properties G_{Ic} and λ , we can now proceed to estimate the expected maximum size of debonds by evaluating G_{TOT} and comparing it with the critical ERR estimated with Equation 3.

Comparison of Fig. 6(a) with Fig. 6(b) shows that the effect of the presence of the 0° layer is to reduce the maximum size of debonds, from $\sim 105^\circ$ in $21 \times 1 - free$ to $\sim 45^\circ$ in $21 \times 1 - 1 \cdot t_{90^\circ}$ for $G_{Ic} = 2 \left[\frac{J}{m^2} \right]$, $\lambda = 0.35$. However, the presence of just 1 “row” of undamaged fibers between the debond and the free surface or the $0^\circ/90^\circ$ interface (in RVEs respectively of UD composites and cross-ply laminates) makes the estimation of the maximum debond size unaffected by the number k of fiber rows (Figures 6(c), 6(d), 6(e) and 6(f)). Moreover, for any given combination of G_{Ic} and λ , the estimated debond size is the same for $n \times k - free$ and $n \times k - 1 \cdot t_{90^\circ}$ for $k \geq 3$. The presence of a debond on the neighboring fiber in the vertical direction (RUCs $21 \times 1 - coupling$ and $21 \times 1 - asymm$) favors instead the growth of larger debonds (Figures 6(g) and 6(h)), with the largest size achieved when debonds are on opposite sides of consecutive fibers ($\sim 115^\circ$ in $21 \times 1 - asymm$ and $\sim 110^\circ$ in $21 \times 1 - coupling$). Considering the different combination of G_{Ic} and λ , it is possible to identify the range of maximum debond propagation: for a 1-fiber row UD composite ($21 \times 1 - free$), $30^\circ - 105^\circ$; for a 1-fiber row 90° ply in a cross-ply laminate ($21 \times 1 - 1 \cdot t_{90^\circ}$), $30^\circ - 45^\circ$; for a k -fiber row UD composite and a k -fiber row 90° ply in a cross-ply laminate with $k \geq 3$ ($21 \times k - free$ and $21 \times k - 1 \cdot t_{90^\circ}$), $40^\circ - 60^\circ$; for consecutive debonds along the vertical direction in an infinite UD composite ($21 \times 1 - coupling$ and $21 \times 1 - asymm$), $80^\circ - 110^\circ$ when debonds are on the same sides and $60^\circ - 115^\circ$ when on opposite sides. These theoretical predictions are in striking agreement with the microscopical observations performed in [2] on $[0_3^0, 90_3^0]_S$ carbon fiber (AS4) - epoxy (8552) specimens subjected to tensile loading (corresponding to RUCs $n \times k - 1 \cdot t_{90^\circ}$ in this article). They report debond sizes in the range $21.4^\circ - 89.2^\circ$ with average value of 49.3° and standard deviation of 11.7° , with 63% of the measurements in the range $40^\circ - 60^\circ$.

5. CONCLUSIONS

Onset and growth of fiber/matrix interface cracks has been studied in Representative Volume Elements (RVEs) of UD composites and $[0_{m \cdot k \cdot 2L}^0, 90_{k \cdot 2L}^0, 0_{m \cdot k \cdot 2L}^0]$ laminates by means of Repeating Unit Cells (RUCs). Analysis of the stress distribution at the fiber/matrix interface in the absence of damage has shown that a stress criterion for initiation would predict, irrespectively of which criterion from those proposed in the literature is chosen, the onset of a debond at 0° or 180° with a semi-aperture $\Delta\theta_0$ in the range $2^\circ - 12^\circ$, corresponding to a margin of 5% on the satisfaction of the criterion. Assuming a mixed-mode criterion for debond propagation and taking into account the previous considerations regarding the stress distribution, the critical Mode I ERR G_{Ic} is estimated to lie in the range $2 - 4.5 \left[\frac{J}{m^2} \right]$. Estimation of maximum debond size finally provides: for a 1-fiber row UD composite, $30^\circ - 105^\circ$; for a 1-fiber row 90° ply in a cross-ply laminate, $30^\circ - 45^\circ$; for a multiple fiber-rows UD composite and a multiple fiber-rows 90° ply in a cross-ply laminate, $40^\circ - 60^\circ$; for consecutive debonds along the vertical direction in an infinite UD composite, $80^\circ - 110^\circ$ when debonds are on the same sides and $60^\circ - 115^\circ$ when on opposite sides. These predictions agree well with the debond size

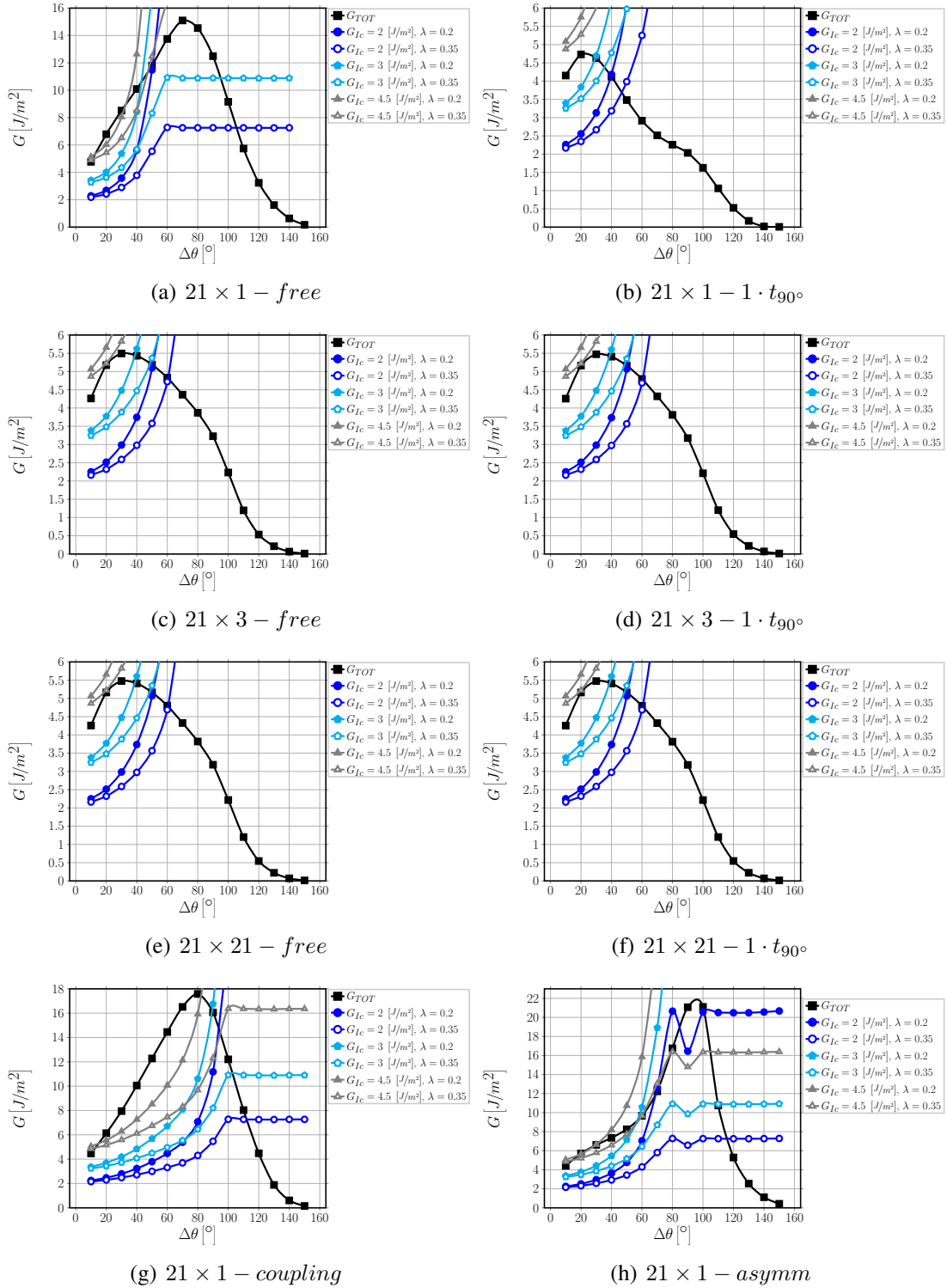


Figure 6. Expected maximum size of debonds, for different RUCs and varying values of G_{IC} and λ .

distribution estimated in [2] through microscopic observations of cross-ply specimens.

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References

- [1] J.W. Hutchinson and Z. Suo. Mixed mode cracking in layered materials. In *Advances in Applied Mechanics*, pages 63–191. Elsevier, 1991.
- [2] Elena Correa, María Inmaculada Valverde, María Luisa Velasco, and Federico París. Microscopical observations of inter-fibre failure under tension. *Composites Science and Technology*, 155:213–220, February 2018.
- [3] Federico París, Elena Correa, and Vladislav Mantič. Kinking of transversal interface cracks between fiber and matrix. *Journal of Applied Mechanics*, 74(4):703, 2007.
- [4] V. Mantič. Interface crack onset at a circular cylindrical inclusion under a remote transverse tension. application of a coupled stress and energy criterion. *International Journal of Solids and Structures*, 46(6):1287–1304, March 2009.
- [5] Luis Pablo Canal, Carlos González, Javier Segurado, and Javier LLorca. Intraply fracture of fiber-reinforced composites: Microscopic mechanisms and modeling. *Composites Science and Technology*, 72(11):1223–1232, June 2012.
- [6] L.E. Asp, L.A. Berglund, and R. Talreja. A criterion for crack initiation in glassy polymers subjected to a composite-like stress state. *Composites Science and Technology*, 56(11):1291–1301, January 1996.
- [7] L.E. Asp, L.A. Berglund, and R. Talreja. Prediction of matrix-initiated transverse failure in polymer composites. *Composites Science and Technology*, 56(9):1089–1097, January 1996.
- [8] P.A. Carraro and M. Quaresimin. A damage based model for crack initiation in unidirectional composites under multiaxial cyclic loading. *Composites Science and Technology*, 99:154–163, July 2014.
- [9] H. Zhang, M.L. Ericson, J. Varna, and L.A. Berglund. Transverse single-fibre test for interfacial debonding in composites: 1. experimental observations. *Composites Part A: Applied Science and Manufacturing*, 28(4):309–315, January 1997.
- [10] Z. Hashin. Analysis of composite materials—a survey. *Journal of Applied Mechanics*, 50(3):481, 1983.

- [11] R.M. Christensen and K.H. Lo. Solutions for effective shear properties in three phase sphere and cylinder models. *Journal of the Mechanics and Physics of Solids*, 27(4):315–330, August 1979.
- [12] Luca Di Stasio, Janis Varna, and Zoubir Ayadi. Growth of interface cracks on consecutive fibers: on the same or on the opposite sides? *Materials Today: Proceedings*, In Press.
- [13] Simulia, Providence, RI, USA. *ABAQUS/Standard User's Manual, Version 6.12*, 2012.
- [14] Ronald Krueger. Virtual crack closure technique: History, approach, and applications. *Applied Mechanics Reviews*, 57(2):109, 2004.
- [15] J. R. Rice. A path independent integral and the approximate analysis of strain concentration by notches and cracks. *Journal of Applied Mechanics*, 35(2):379, 1968.
- [16] C. Sandino, E. Correa, and F. París. Numerical analysis of the influence of a nearby fibre on the interface crack growth in composites under transverse tensile load. *Engineering Fracture Mechanics*, 168:58–75, December 2016.
- [17] Luca Di Stasio, Janis Varna, and Zoubir Ayadi. Energy release rate of the fiber/matrix interface crack in UD composites under transverse loading: effect of the fiber volume fraction and of the distance to the free surface and to non-adjacent debonds. *Theoretical and Applied Fracture Mechanics*, 103, October 2019.
- [18] P.A. Carraro and M. Quaresimin. Modeling the crack initiation in unidirectional laminates under multiaxial fatigue loading1. In *Modeling Damage, Fatigue and Failure of Composite Materials*, pages 357–375. Elsevier, 2016.
- [19] Leif E. Asp, Lars A. Berglund, and Peter Gudmundson. Effects of a composite-like stress state on the fracture of epoxies. *Composites Science and Technology*, 53(1):27–37, January 1995.