

# Energy release rate of the fiber/matrix interface crack in cross-ply $[0_{m \cdot 2n}^{\circ}, 90_n^{\circ}]_S$ laminates under transverse loading: debond/bi-material interface interaction

Luca Di Stasio<sup>a,b</sup>, Janis Varna<sup>b</sup>, Zoubir Ayadi<sup>a</sup>

<sup>a</sup> Université de Lorraine, EEIGM, IJL, 6 Rue Bastien Lepage, F-54010 Nancy, France

<sup>b</sup> Luleå University of Technology, University Campus, SE-97187 Luleå, Sweden

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## Abstract

The effects of crack shielding, fiber content and ratio of  $0^{\circ}$  to  $90^{\circ}$  ply thickness on fiber/matrix debond growth in thin cross-ply laminates are investigated with Representative Volume Elements (RVEs) of different ordered microstructures. Debond growth is characterized by the estimation of the Energy Release Rates (ERRs) using the Virtual Crack Closure Technique (VCCT) and the J-integral. It is found that

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## 1. Introduction

Since the development of the *spread tow* technology or “FUKUI method” [1, 2], significant efforts have been directed toward the characterization of *thin-ply* laminates [3, 4, 5, 6, 7, 8, 9, 10, 11, 12, 13, 14, 15] and their application to mission-critical structures in the aerospace sector [16, 17, 18, 19].

At the lamina level, the use of *thin-ply*s leads to more regular and homogeneous microstructures but no significant improvement in static properties except for an apparent improvement in compressive strength [12]. Improvements in fatigue life have been observed, although contrasting results can be found in the literature [4, 5, 6]. The beneficial effect of the use of *thin-ply*s with respect to damage propagation has been instead commonly observed by different re-

searchers under static [3, 6, 7, 8, 9, 10, 11, 12], fatigue [4, 6, 7, 8, 12] and impact loadings [6, 7, 8, 12]. It seems apparent that *thin-ply* laminates possess an increased ability to delay, and in some cases even suppress, the onset and  
15 propagation of transverse cracks (or matrix or micro-cracks).

The first appearance of transverse cracking phenomena is known to be characterized by the appearance of fiber/matrix interface cracks (also referred to as debonds), which grow along the fiber's arc direction, then kink out of the interface and coalesce forming a transverse crack [20]. Different approaches  
20 have been applied to model the initiation and growth of debonds. The Cohesive Zone Model (CZM) has been used to mimic the propagation of debonds along fiber interfaces; coupled with a failure criterion for the matrix, it has provided simulations of the growth of transverse cracks starting from a virgin material [21, 22, 23, 24]. The main advantages of this approach are the pos-  
25 sibility to observe the development of a simulated crack path and to record a load-displacement curve to compare with experimental measurement. However, various observations cast a doubt about the applicability of the CZM: the bi- (for 2D models) and tri- (in 3D) axiality of the matrix stress state in the inter-fiber region that is linked with a cavitation-like failure of the polymer [25];  
30 the locality and mode dependency of the interface failure [26]; the problematic use at the microscopic level of properties measured in UD specimens at the laminate level [22]. A second approach that obviates these drawbacks is the application of Linear Elastic Fracture Mechanics (LEFM) arguments to the study of debond growth. The analysis focuses on the evaluation of Mode I and Mode  
35 II Energy Release Rate (ERR) at the crack tip by means of the Virtual Crack Closure Technique (VCCT) [27] or the J-Integral method [28]. The stress and strain field, required for the ERR computation, can be solved by application of different methodologies such as analytical solutions [29], the Boundary Element Method (BEM) [30] or the Finite Element Method (FEM) [31]. Different  
40 works have followed this approach and studied models of one or two fibers in an effectively infinite matrix [32, 33, 34, 35, 36] and of an hexagonal cluster of fibers in an effectively infinite homogenized UD composite [37, 31]. The problem

of debond growth along the fiber-matrix interface in a cross-ply laminate has been only addressed very recently in [38, 39], where the author embed a single  
45 partially debonded fiber in an effectively infinite homogenized  $90^\circ$  ply bounded by homogenized  $0^\circ$  layers. Thus, the effect of debond-debond interaction and of the relative proximity of a bi-material interface on the debond's ERR in cross-ply laminates is yet to be addressed. The present work is devoted to this problem. Models of Repeating Unit Cells (RUCs) are developed to represent  
50 laminates with different degrees of damage (here only in the form of debonds). The number of fully bonded fibers across the thickness of the  $90^\circ$  ply is varied in order to investigate the effect of the proximity of the bi-material interface. The thickness of the bounding  $0^\circ$  layers is also analyzed as a parameter of the study. The stress and strain fields are solved with the Finite Element Method  
55 in Abaqus [40] and the crack characterized by its Mode I and Mode II ERR, calculated with the VCCT and the J-integral method.

## 2. RVE models & FE discretization

### 2.1. Introduction & Nomenclature

In the present work, we investigate debond development in cross-ply  $[0_{m \cdot 2n}^\circ, 90_n^\circ]_S$   
60 laminates under in-plane transverse tension. The interaction between debonds in the presence of a stiff bi-material interface is studied with the use of different RUCs (see Figures 1 and 2 in Sec. 2.2), in which only the central fiber presents damage in the form of a debond. Repetition of the composite RUC can occur only along the in-plane transverse direction only, thus representing a cross-ply  
65 laminate with a thin or even ultra-thin  $90^\circ$  ply in the middle.

The thickness of the  $90^\circ$  ply depends on the number of fibers present across the thickness (the vertical or  $z$  direction in Figures 1 and 2) and the value of the fiber volume fraction  $V_f$ . On the other hand, the thickness of the  $0^\circ$  layers can be assigned freely as a multiple of the  $90^\circ$  ply thickness, i.e.  $t_{0^\circ} = i \cdot t_{90^\circ}$  where  
70  $i$  is an arbitrary integer. The thickness ratio  $i$  could in theory be assumed to be a real positive number; however, it seems more reasonable to consider it only

as a positive integer based on practical considerations on the actual manufacturing of laminates (stacking of a discrete number of pre-impregnated layers). Thus, the thickness ratio  $i$  represents one additional parameter for the investigation. In the RUCs proposed, we consider the  $90^\circ$  ply with debonds as a series of stacked damaged and undamaged fiber rows, each row with only one fiber in the thickness direction. All the RUCs present regular microstructures with fibers placed according to a square-packing configuration and consequently they are Representative Volume Elements (RVE) of cross-ply laminates with a certain distribution of debonds in the middle  $90^\circ$  layer. In the following, let us consider in-plane coordinates  $x$  and  $y$ , where  $x$  is in the transverse direction of the cross-ply laminate under consideration. In the presence of a load in the  $x$ -direction, the strain in the  $y$ -direction is small, due to the very small minor Poisson's ratio of the laminate. Furthermore, debonds are considered to be significantly longer in the fiber direction than in the arc direction [41]. Therefore we use 2D models under the assumption of plane strain, defined in the  $x - z$  section of the composite. The study presented in this paper thus applies to long debonds and its focus is on understanding the mechanisms of growth along their arc direction. The laminates are assumed to be subject to transverse tensile strain, which is applied in the form of a constant displacement in the  $x$ -direction along both vertical boundaries of the RUC as shown in Figure 3.

In summary, the models are differentiated by: first, the spacing between debonds along the horizontal direction in the  $90^\circ$  layer, which corresponds to the number  $n$  of fibers in the RUC's horizontal direction; second, the thickness of the middle  $90^\circ$  ply measured in terms of the number  $k$  of fiber rows; third, the factor  $i$  which provides the thickness of the  $0^\circ$  layers as an integer multiple of the  $90^\circ$  ply thickness. It thus seems natural to introduce the common notation  $n \times k - i \cdot t_{90^\circ}$ . A final additional model is considered to study the effect of equivalent boundary conditions 3.4. This final model is constituted by only one partially debonded fiber. The application of coupling of horizontal displacements in the form of a constant applied displacement along the right and left sides allows for repetition along the horizontal direction. The pres-

ence of coupling of vertical displacements and a linear distribution of horizontal displacements on the bottom and top surfaces models the presence of the stiff  
105 bi-material interface between the  $90^\circ$  and the  $0^\circ$  layers. This model is referred to as  $1 \times 1 - H + V$  given that: it has respectively 1 fiber in the horizontal and in the vertical direction; on the top and bottom surfaces, both horizontal (H) and vertical (V) displacements are assigned.

## 2.2. Models of Representative Volume Element (RVE)

110 The first family of models is represented in Figure 1. It represents a set of  $[0_{m \cdot 2n}^\circ, 90_n^\circ]_S$  cross-ply laminates with an ultra-thin  $90^\circ$  layer, constituted by a single row of fibers across the thickness. Debonds appear at regular intervals measured in terms of number  $n$  of fully bonded fibers present between them, which in turn correspond to the number of fibers along the horizontal direction  
115 of the RUC as highlighted in Fig. 1. They are thus the  $n \times 1 - i \cdot t_{90^\circ}$  models, where  $i = 1, 10$  and  $n$  is an integer  $\geq 1$  ( $n = 1$  corresponds to the case of a debond appearing on all the fibers in the central  $90^\circ$  layer). These models are quite extreme, but allow to focus on the interaction between debonds and the inter-ply bi-material interface. Furthermore, the *spread tow* technology is today  
120 capable of producing cross-ply laminates with the central  $90^\circ$  layer thickness only 4 – 5 times the fiber diameter, as shown for example in [9], which give practical relevance even to such extreme models.

The second set of models considers instead cross-ply laminates with a central  $90^\circ$  ply of variable thickness, measured in terms of number  $k$  of fiber rows  
125 appearing in the vertical direction in Figure 2. Once again, debonds appear at regular intervals measured in terms of number  $n$  of fully bonded fibers present between them, which in turn correspond to the number of fibers along the horizontal direction of the RUC as highlighted in Fig. 2. These models are thus the  $n \times k - i \cdot t_{90^\circ}$  models, where  $i = 1, 10$ ,  $k > 1$  and  $n$  is an integer  $\geq 1$  ( $n = 1$   
130 corresponds to the case of a debond appearing on all the fibers of the central fiber row in the  $90^\circ$  layer).

By increasing the number  $n$  of fibers in the horizontal direction in the RUC,

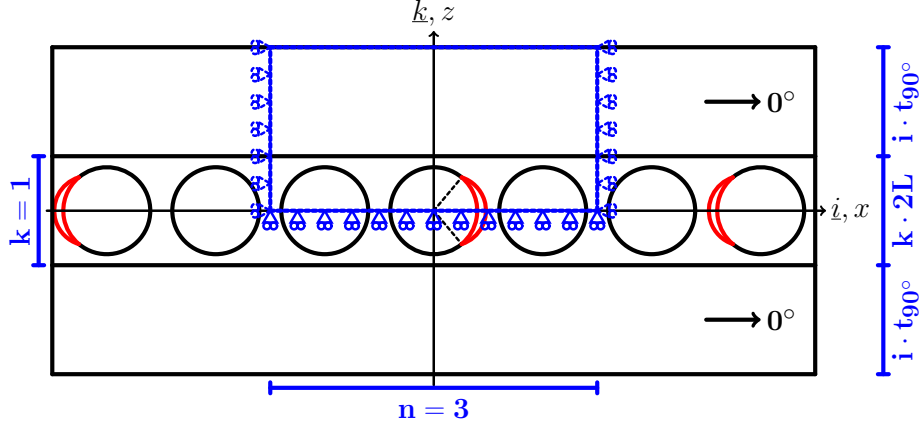


Figure 1: Models of  $[0_m^{\circ}, 2n, 90_n^{\circ}]_S$  cross-ply laminates with an ultra-thin  $90^{\circ}$  layer, where the  $90^{\circ}$  ply is made up by a single “row” of fibers. Debonds are repeating at different distances, measured in terms of the number  $n$  of fully bonded fibers appearing between two consecutive debonds.

decreasing levels of damage (debonds spaced further apart) are considered to be present in the laminate. By increasing the number  $k$  of fiber rows, the thickness  
 135 of the  $90^{\circ}$  layer is increased and the effect of the relative proximity of the interply bi-material interface can thus be studied. Finally, by increasing the factor  $i$ , the thickness of the  $0^{\circ}$  layers is increased for a given thickness of the  $90^{\circ}$ , which allows the investigation of the size effect or *in-situ* effect for the fiber-matrix interface crack.

### 140 2.3. Finite Element (FE) discretization

The RUCs are discretized using the Finite Element Method (FEM) with the commercial FEM package Abaqus [40]. The length  $l$  and height  $h$  of the model are determined by the number of fibers  $n$  in the horizontal direction, the number of fiber rows  $k$  across the thickness and the thickness ratio  $i$  (see  
 145 Sec. 2.2) according to Eq. 1:

$$l = 2nL \quad h = (1 + 2i)kL. \quad (1)$$

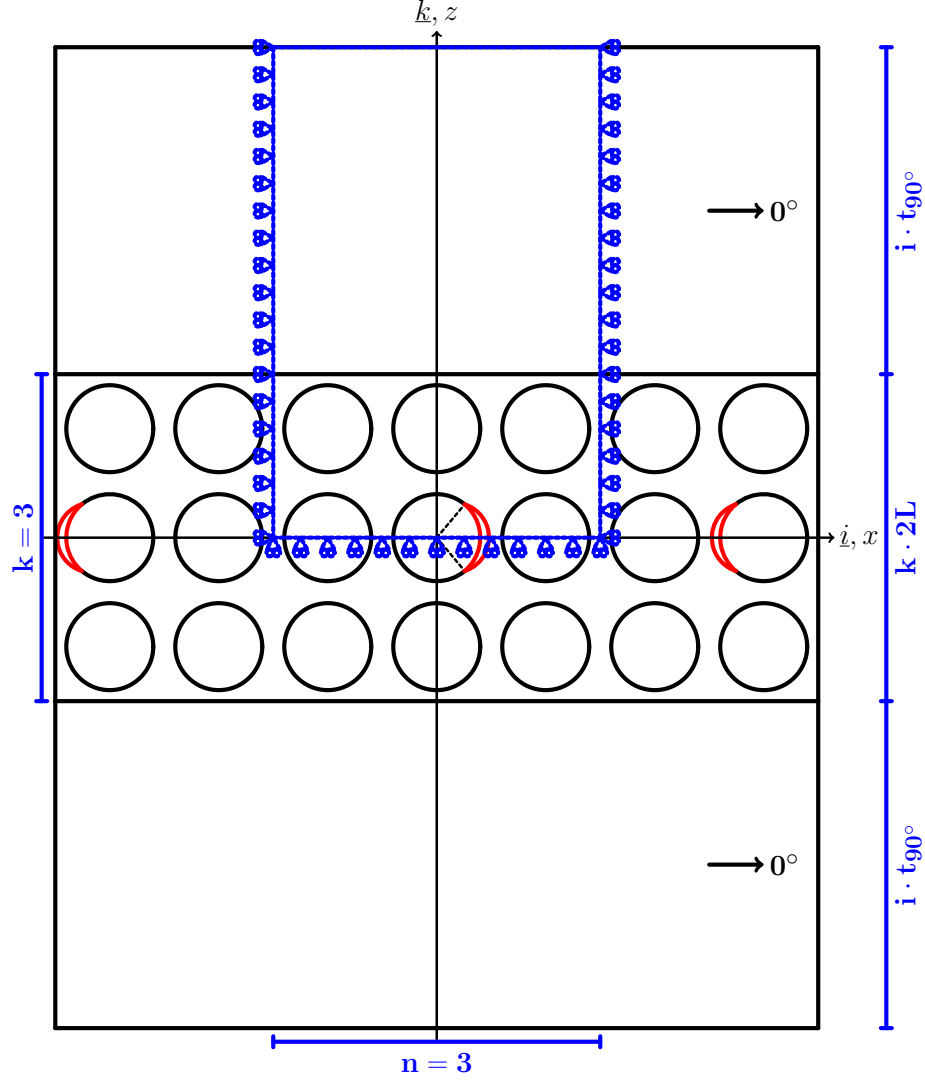


Figure 2: Models of  $[0_m^{\circ} \cdot 2n, 90_n^{\circ}]_S$  cross-ply laminates with a  $90^{\circ}$  layer of variable thickness, determined by the number  $k$  of “rows” of fibers along the vertical direction. Debonds are repeating at different distances along the horizontal direction, measured in terms of the number  $n$  of fully bonded fibers appearing between two consecutive debonds.

In Eq. 1,  $2L$  is the length of a one-fiber unit (see Fig. 3), which in turn is as a function of the fiber volume fraction  $V_f$  and the fiber radius according to

$$L = \frac{R_f}{2} \sqrt{\frac{\pi}{V_f}}. \quad (2)$$

Each fiber in the model has the same radius  $R_f$ , equal to  $1 \mu m$ . This specific value has no physical meaning per se and it has been selected for simplicity. It is useful to observe that, in a linear elastic solution as the one described in the present article, the ERR is proportional to the geometrical dimensions of the model and thus re-evaluation of the ERR for fibers of any size requires just a multiplication. Furthermore, the local and global  $V_f$  are everywhere equal thanks to the relationships in Eqs. 1 and 2.

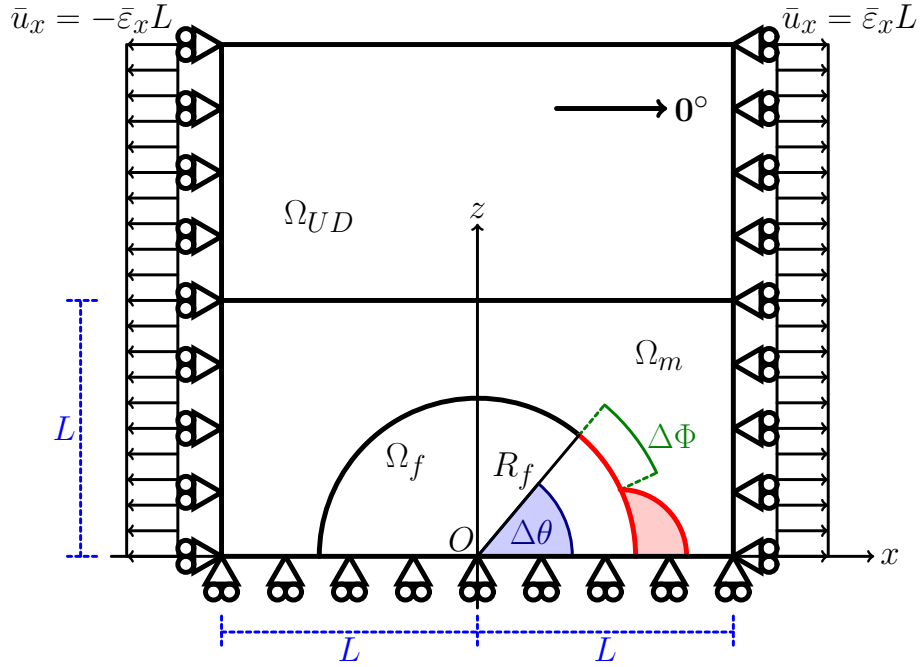


Figure 3: Schematic of the model with its main parameters.

The debond appears symmetrically with respect to the  $x$  axis (see Fig. 3) and we characterize it with the angular size  $\Delta\theta$  (the full debond size is thus  $2\Delta\theta$ ). In the case of large debond sizes ( $\geq 60^\circ - 80^\circ$ ), a region of size  $\Delta\Phi$  to be determined by the solution itself appears at the crack tip. In this region, called the *contact zone*, the crack faces are in contact and slide on each other. Due



160 to existence of the contact zone, frictionless contact is considered between the  
two crack faces to avoid interpenetration and allow free sliding. Symmetry with  
respect to the  $x$  axis is applied on the lower boundary. The upper boundary  
is free, except for the model  $1 \times 1 - H + V$  which requires on the upper side  
kinematic coupling of vertical displacements and applied linearly distributed  
165 horizontal displacements. Kinematic coupling on the  $x$ -displacement is applied  
along the left and right boundaries of the model in the form of a constant  $x$ -  
displacement  $\pm \bar{\varepsilon}_x l$ , corresponding to transverse strain  $\bar{\varepsilon}_x$  equal to 1%.

Table 1: Summary of the mechanical properties of fiber, matrix and UD layer.  $E$  stands  
for Young's modulus,  $\mu$  for shear modulus and  $\nu$  for Poisson's ratio. Indexes  $L$  and  $T$  stand  
respectively for *longitudinal* and *transverse*.

<b>Material</b>	$V_f$ [%]	$E_L$ [GPa]	$E_T$ [GPa]	$\mu_{LT}$ [GPa]	$\nu_{LT}$ [-]	$\nu_{TT}$ [-]
Glass fiber	-	70.0	70.0	29.2	0.2	0.2
Epoxy	-	3.5	3.5	1.25	0.4	0.4
UD	60.0					

### 3. Results & Discussion

- 3.1. *Interaction between debonds in a  $90^\circ$  ply with a single layer of fibers inside*  
170 *a  $[0_n^\circ, 90^\circ]_S$  laminate*
- 3.2. *Interaction between layers of fully bonded fibers and a centrally located line*  
*of debonded fibers in a  $90^\circ$  ply inside a  $[0_n^\circ, 90^\circ]_S$  laminate*
- 3.3. *Interaction of debonds within a  $90^\circ$  ply with multiple layers of fibers inside*  
*a  $[0_n^\circ, 90^\circ]_S$  laminate*
- 175 3.4. *Comparison with the single fiber model with equivalent boundary conditions*

### 4. Conclusions & Outlook

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