

NOTE: FOR SUPERCRITICAL AIRFOILS USE  $\Delta M_{CR} = 0.05$

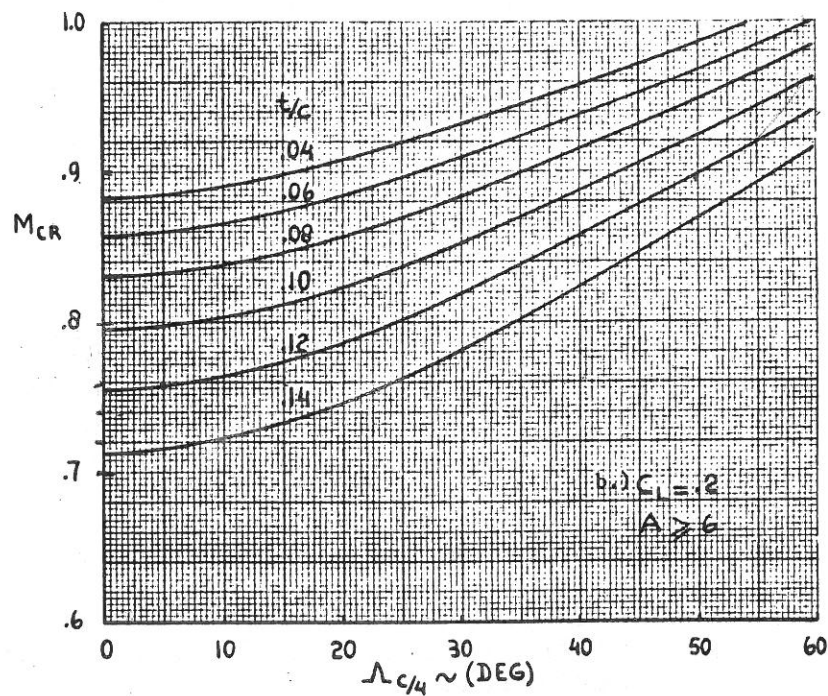
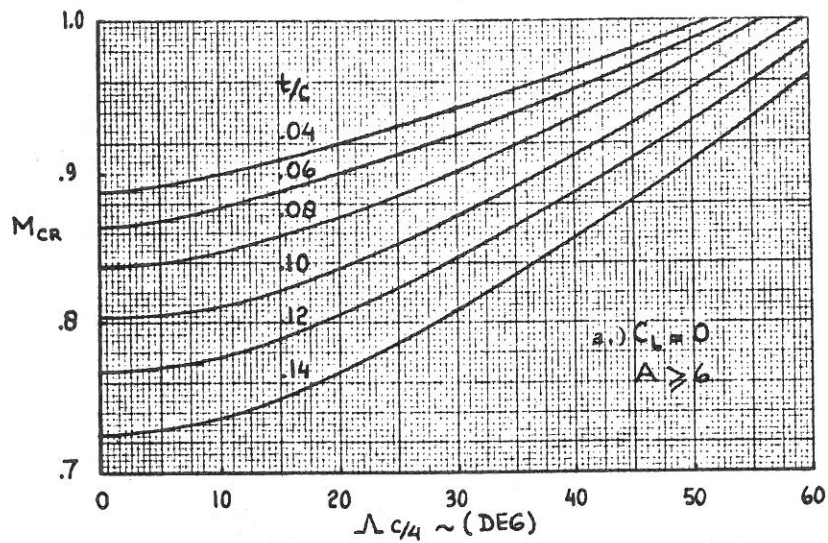


Figure 6.1a Effect of Thickness Ratio and Sweep Angle on Critical Mach Number