Table 1: Spanwise distribution of bound circulation & total wing lift coefficient for all wing configurations for N = 128

			Rectangular			Tapered			Elliptical	
i	YC		Γ	Cl/CL		Γ	Cl/CL		Γ	Cl/CL
1	0.000		0.051	1.390		0.065	1.068		0.061	1.201
2	0.012		0.051	1.389		0.065	1.076		0.060	1.200
3	0.025		0.051	1.386		0.065	1.083		0.060	1.198
4	0.037		0.051	1.383		0.064	1.088		0.060	1.196
5	0.049		0.051	1.379		0.064	1.093		0.060	1.193
6	0.062		0.051	1.375		0.064	1.097		0.060	1.190
7	0.074		0.051	1.370		0.063	1.100		0.060	1.186
8	0.086		0.051	1.364		0.063	1.103		0.059	1.182
9	0.098		0.050	1.358		0.063	1.105		0.059	1.177
10	0.111		0.050	1.352		0.062	1.107		0.059	1.173
11	0.123		0.050	1.345		0.062	1.108		0.058	1.168
12	0.135		0.050	1.338		0.061	1.109		0.058	1.163
13	0.147		0.049	1.331		0.061	1.109		0.058	1.157
14	0.159		0.049	1.324		0.060	1.109		0.057	1.152
15	0.172		0.049	1.316		0.059	1.109		0.057	1.147
16	0.184		0.048	1.308		0.059	1.108		0.057	1.141
17	0.196		0.048	1.300		0.058	1.107		0.056	1.135
18	0.208		0.048	1.292		0.058	1.106		0.056	1.129
19	0.220		0.048	1.284		0.057	1.105		0.055	1.123
20	0.232		0.047	1.275		0.056	1.103		0.055	1.117
21	0.244		0.047	1.267		0.056	1.101		0.054	1.111
22	0.256		0.047	1.258		0.055	1.099		0.054	1.105
23	0.268		0.046	1.249		0.054	1.097		0.053	1.099
24	0.280		0.046	1.240		0.054	1.095		0.053	1.092
25	0.291		0.046	1.231		0.053	1.092		0.052	1.086

26	0.303		0.045	1.222		0.052	1.089		0.052	1.080
27	0.315		0.045	1.213		0.051	1.086		0.051	1.073
28	0.327		0.045	1.204		0.051	1.083		0.051	1.067
29	0.338		0.044	1.194		0.050	1.079		0.050	1.060
30	0.350		0.044	1.185		0.049	1.076		0.050	1.054
31	0.361		0.044	1.175		0.049	1.072		0.049	1.047
32	0.373		0.043	1.166		0.048	1.068		0.049	1.041
33	0.384		0.043	1.156		0.047	1.064		0.048	1.034
34	0.395		0.042	1.147		0.047	1.060		0.048	1.027
35	0.407		0.042	1.137		0.046	1.056		0.047	1.021
36	0.418		0.042	1.127		0.045	1.052		0.046	1.014
37	0.429		0.041	1.118		0.045	1.048		0.046	1.008
38	0.440		0.041	1.108		0.044	1.043		0.045	1.001
39	0.451		0.041	1.098		0.043	1.038		0.045	0.995
40	0.462		0.040	1.088		0.043	1.034		0.044	0.988
41	0.473		0.040	1.079		0.042	1.029		0.044	0.982
42	0.484		0.040	1.069		0.041	1.024		0.043	0.975
43	0.495		0.039	1.059		0.041	1.019		0.042	0.969
44	0.505		0.039	1.049		0.040	1.014		0.042	0.962
45	0.516		0.038	1.039		0.039	1.009		0.041	0.956
46	0.526		0.038	1.029		0.039	1.004		0.041	0.949
47	0.537		0.038	1.019		0.038	0.999		0.040	0.943
48	0.547		0.037	1.009		0.037	0.993		0.040	0.937
49	0.557		0.037	1.000		0.037	0.988		0.039	0.931
50	0.568		0.037	0.990		0.036	0.983		0.038	0.924
51	0.578		0.036	0.980		0.035	0.977		0.038	0.918
52	0.588		0.036	0.970		0.035	0.972		0.037	0.912
53	0.598		0.036	0.960		0.034	0.966		0.037	0.906
54	0.608		0.035	0.950		0.034	0.960		0.036	0.900

55	0.617		0.035	0.940		0.033	0.955		0.035	0.894
56	0.627		0.034	0.930		0.032	0.949		0.035	0.888
57	0.636		0.034	0.920		0.032	0.943		0.034	0.882
58	0.646		0.034	0.910		0.031	0.937		0.034	0.876
59	0.655		0.033	0.900		0.031	0.931		0.033	0.870
60	0.665		0.033	0.890		0.030	0.925		0.033	0.864
61	0.674		0.033	0.880		0.030	0.919		0.032	0.858
62	0.683		0.032	0.870		0.029	0.913		0.031	0.853
63	0.692		0.032	0.860		0.028	0.907		0.031	0.847
64	0.701		0.031	0.850		0.028	0.901		0.030	0.842
65	0.709		0.031	0.840		0.027	0.895		0.030	0.836
66	0.718		0.031	0.830		0.027	0.889		0.029	0.831
67	0.726		0.030	0.820		0.026	0.882		0.029	0.825
68	0.735		0.030	0.810		0.026	0.876		0.028	0.820
69	0.743		0.030	0.800		0.025	0.870		0.027	0.815
70	0.751		0.029	0.790		0.025	0.863		0.027	0.810
71	0.759		0.029	0.780		0.024	0.857		0.026	0.805
72	0.767		0.029	0.770		0.024	0.850		0.026	0.799
73	0.775		0.028	0.760		0.023	0.844		0.025	0.795
74	0.783		0.028	0.750		0.023	0.837		0.025	0.790
75	0.791		0.027	0.739		0.023	0.830		0.024	0.785
76	0.798		0.027	0.729		0.022	0.823		0.024	0.780
77	0.805		0.027	0.719		0.022	0.817		0.023	0.775
78	0.813		0.026	0.708		0.021	0.810		0.023	0.771
79	0.820		0.026	0.698		0.021	0.803		0.022	0.766
80	0.827		0.025	0.688		0.020	0.795		0.022	0.762
81	0.834		0.025	0.677		0.020	0.788		0.021	0.758
82	0.840		0.025	0.667		0.020	0.781		0.021	0.753
83	0.847		0.024	0.656		0.019	0.773		0.020	0.749

84	0.853		0.024	0.645		0.019	0.765		0.020	0.745
85	0.860		0.024	0.635		0.018	0.758		0.019	0.741
86	0.866		0.023	0.624		0.018	0.750		0.019	0.737
87	0.872		0.023	0.613		0.018	0.742		0.018	0.733
88	0.878		0.022	0.602		0.017	0.733		0.018	0.730
89	0.884		0.022	0.591		0.017	0.725		0.017	0.726
90	0.890		0.021	0.580		0.016	0.716		0.017	0.722
91	0.895		0.021	0.568		0.016	0.707		0.016	0.719
92	0.901		0.021	0.557		0.016	0.698		0.016	0.716
93	0.906		0.020	0.545		0.015	0.689		0.015	0.712
94	0.911		0.020	0.534		0.015	0.679		0.015	0.709
95	0.916		0.019	0.522		0.015	0.669		0.014	0.706
96	0.921		0.019	0.510		0.014	0.659		0.014	0.703
97	0.926		0.018	0.498		0.014	0.648		0.013	0.700
98	0.930		0.018	0.486		0.014	0.637		0.013	0.697
99	0.935		0.018	0.473		0.013	0.626		0.012	0.695
100	0.939		0.017	0.461		0.013	0.614		0.012	0.692
101	0.943		0.017	0.448		0.012	0.602		0.012	0.690
102	0.947		0.016	0.435		0.012	0.590		0.011	0.687
103	0.951		0.016	0.422		0.012	0.577		0.011	0.685
104	0.955		0.015	0.408		0.011	0.563		0.010	0.683
105	0.958		0.015	0.395		0.011	0.549		0.010	0.681
106	0.962		0.014	0.381		0.011	0.534		0.009	0.679
107	0.965		0.014	0.367		0.010	0.519		0.009	0.678
108	0.968		0.013	0.353		0.010	0.503		0.009	0.676
109	0.971		0.013	0.339		0.009	0.487		0.008	0.675
110	0.974		0.012	0.324		0.009	0.470		0.008	0.674
111	0.977		0.011	0.309		0.009	0.452		0.007	0.673
112	0.979		0.011	0.294		0.008	0.433		0.007	0.672

113	0.982		0.010	0.279		0.008	0.414		0.006	0.671
114	0.984		0.010	0.263		0.007	0.394		0.006	0.671
115	0.986		0.009	0.247		0.007	0.373		0.006	0.671
116	0.988		0.009	0.231		0.007	0.352		0.005	0.671
117	0.990		0.008	0.215		0.006	0.329		0.005	0.672
118	0.992		0.007	0.198		0.006	0.306		0.004	0.673
119	0.993		0.007	0.181		0.005	0.282		0.004	0.675
120	0.995		0.006	0.164		0.005	0.257		0.004	0.678
121	0.996		0.005	0.147		0.004	0.232		0.003	0.682
122	0.997		0.005	0.130		0.004	0.205		0.003	0.688
123	0.998		0.004	0.112		0.003	0.178		0.002	0.696
124	0.998		0.003	0.094		0.003	0.150		0.002	0.708
125	0.999		0.003	0.076		0.002	0.122		0.002	0.728
126	1.000		0.002	0.058		0.002	0.093		0.001	0.764
127	1.000		0.001	0.040		0.001	0.065		0.001	0.843
128	1.000		0.001	0.023		0.001	0.037		0.000	1.083

Table 2: Total wing lift coefficient (C_L), total wing induced drag coefficient (C_{Di}). & δ for a Rectangular Wing

Rectangular Wing			
N	C_L	C_Di	δ
2	0.3528	0.0036	-0.2777
8	0.3033	0.0034	-0.0600
32	0.2975	0.0035	-0.0056
128	0.2963	0.0035	0.0081

Table 3: Total wing lift coefficient (C_L), total wing induced drag coefficient (C_{Di}). & δ for a Tapered Wing

Tapered Wing			
N	C_L	C_Di	δ
2	0.4222	0.0060	-0.1508
8	0.3242	0.0044	0.0520
32	0.3174	0.0044	0.1025
128	0.3163	0.0044	0.1165

Table 4: Total wing lift coefficient (C_L), total wing induced drag coefficient (C_{Di}). & δ for a Elliptical Wing

Elliptical Wing			
N	C_L	C_Di	δ
2	0.3895	0.0047	-0.2241
8	0.3234	0.0041	-0.0065
32	0.3177	0.0042	0.0513
128	0.3167	0.0043	0.0658