

Name: \_\_\_\_\_ Score: \_\_\_\_\_

## 38 Written questions

Term

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For isothermal flow, pressure is proportional to what power of density?

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Term

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How do static values of pressure, temperature and density vary when the gas undergoes compression, such as at a stagnation point? they all go \_\_\_\_\_.

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The critical pressure ratio required to achieve sonic condition is

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Carl de Laval of Sweden was the first to make industrial use of converging-diverging \_\_\_\_\_ to achieve much higher RPMs in a steam turbine.

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Is the Energy Equation very important in the study of compressible flows?

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On physical grounds, at the same temperature, is the speed of sound higher in water than in air?

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In an \_\_\_\_\_ flow, the total temperature remains constant.

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Higher the flow Mach number ( $M_1$ ) ahead of a normal shock, \_\_\_\_\_ the Mach number behind the shock/

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Chuck Yeager's historic first \_\_\_\_\_ flight took place in the year 1947, only two years after the end of WWII.

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Speed of sound in a gas is the speed of propagation of \_\_\_\_\_ pressure disturbances through the gas.

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Prandtl relation can be written as: \_\_\_\_\_; where  $a^*$  represents the speed of sound computed from flow being at \_\_\_\_\_ conditions.

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For isentropic flow, pressure is proportional to what power of density (give the symbol)?

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A \_\_\_\_\_ perfect gas is one for which,  $C_p(T)$ ,  $C_v(T)$  are both function of  $T$  and not constant.

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How does entropy vary across a shock wave?

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What type of pressure remains constant in an isentropic flow?

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T or F: When the flow Mach number ahead of a normal shock is increased infinitely, the flow density behind the shock also reaches infinity.

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Give the equation for the speed of sound in a gas:

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Is the propagation of sound speed in air as an isentropic process?

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Can we analytically solve the theta-beta-M relation for beta?

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A \_\_\_\_\_ perfect gas is one for which:  $c_p$  and  $c_v$  are constant.

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Rankin-Hugoniot relation across a normal shock gives us the pressure jump  $p_2/p_1$  as function of (Mach number,  $M_1$ ; or density ratio  $\rho_2/\rho_1$  across the shock; or entropy jump,  $s_2/s_1$ ).

Select one

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\_\_\_\_\_ flow refers to a constant-entropy flow.

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Has the shock wave system of an aircraft in flight ever been recorded?

**Term**

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The \_\_\_\_\_ law of thermodynamics allows us to predict the direction that physical processes take.

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In an adiabatic flow, the total \_\_\_\_\_ remains constant.

**Term**

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There is a portion of any curved shock wave that is \_\_\_\_\_ to the stagnation streamline.

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We can not have an expansion shock (i.e.,  $p_2 < p_1$ ), because this would violate the \_\_\_\_\_ law of thermodynamics. The critical pressure ratio required to achieve sonic condition is \_\_\_\_\_.

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Give the equation for calculating the Mach angle  $\mu$ :

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Tor F: The stagnation pressure across a normal shock (i.e.,  $p_{02}$ ) is higher than the stagnation pressure in front of the shock ( $p_{01}$ ). This variation is the same as static pressure jump (i.e.,  $p_2 > p_1$ ).

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We have a \_\_\_\_\_ perfect gas for  $T < 1000$  K.

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At very high temperature where dissociation and ionization degree of freedom are important, then air's specific heat ratio  $\gamma = C_p/C_v$  would be: increased, decreased, unchanged.  
( $\gamma < 1.4$ ;  $\gamma > 1.4$ , always equal to 1.4)

(Select one)

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The critical \_\_\_\_\_ ratio required to achieve sonic condition is \_\_\_\_\_ (Mach # = 1).

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**Term**

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The Mach number downstream of a/an \_\_\_\_\_ shock wave is always subsonic.

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In \_\_\_\_\_, the total stagnation pressure remains constant.

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What type of pressure jump jumps up across a shock wave?

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A/an \_\_\_\_\_ and reversible process is called isentropic.

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**Term**

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Does the second law of thermodynamics allow the process where the entropy for the system and surroundings decrease?

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A curved shock wave is also called a \_\_ shock wave, usually created by blunt object in supersonic flight, such as the Space Shuttle Orbiter.

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