

# Improved Lunar Lander Handling Qualities Through Control Response Type and Display Enhancements

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A piloted simulation that studied the handling qualities for a precision lunar landing task from final approach to touchdown is presented. A vehicle model based on NASA's Altair lunar lander was used to study the combinations of factors that provide satisfactory pilot-vehicle performance and workload; control and propulsion system details not available for Altair were derived from Apollo lunar module data. Eight Space Shuttle and Apollo astronauts and three NASA test pilots served as evaluation pilots, providing Cooper-Harper ratings, Task Load Index ratings, and qualitative comments. Pilots flew seven combinations of control response types along with two varieties of guidance and a nonguided approach. The response types included the rate command with attitude hold system used on the Apollo lunar module, a new velocity increment command response type, and three response types designed for precise horizontal maneuvering during terminal descent. It was found that velocity increment command improved handling qualities when compared with the baseline Apollo design, receiving predominantly Level 1 ratings. This response type could be flown without guidance cues, which was very difficult in the Apollo system design, and resulted in approximately equivalent touchdown accuracies and propellant burn. The terminal descent response types did not improve handling qualities.

## Nomenclature

$c_{LQ}$	=	linear-quadratic shaping parameter
$g_{\text{lunar}}$	=	gravitational acceleration at lunar surface, $\text{ft}/\text{s}^2$
$h$	=	height above lunar surface, ft
$\dot{h}$	=	time derivative of $h$ , $\text{ft}/\text{s}$
$I_0$	=	moment of inertia, slug- $\text{ft}^2$
$k$	=	parameter for tradeoff between propellant consumption and error settling time
$M$	=	moment about a single axis, $\text{ft} \cdot \text{lb}_f$
$m$	=	vehicle mass, slug
$p$	=	roll rate, deg/s
$q$	=	pitch rate, deg/s
$r$	=	yaw rate, deg/s
$T$	=	thrust of descent engine, $\text{lb}_f$
$T^o$	=	value of $T$ required for vertical force equilibrium, $\text{lb}_f$
$t$	=	time, s
$\alpha$	=	angular acceleration about a single axis, $\text{deg}/\text{s}^2$
$\delta$	=	normalized displacement of rotation hand controller
$\theta$	=	Euler pitch angle, deg
$\tau$	=	time constant, s
$\varphi$	=	Euler roll angle, deg
$\psi$	=	Euler yaw angle, deg
$\omega$	=	commanded angular rate, deg/s

## Superscripts and Subscripts

body	=	with respect to body axes
cmd	=	commanded value
$D$	=	down component
DB	=	dead band
DE	=	descent engine
$E$	=	east component
err	=	error value
flat	=	offset of inner switching curve from deadband value
HI	=	boundary of rate error for two-jet vs four-jet control
max	=	maximum command
$N$	=	north component
$P$	=	pitch axis
$R$	=	roll axis
ROD	=	rate of descent
$Y$	=	yaw axis
*	=	reference variable

## Introduction

HANDLING qualities are those characteristics of a flight vehicle that govern the ease and precision with which a pilot is able to perform a flying task [1]. The way in which particular vehicle factors affect handling qualities has been studied in aircraft for decades [2–4], and reference standards for the handling qualities of both fixed-wing aircraft [5] and rotary-wing aircraft [6] have been developed and are now in common use. These standards define a subset of the dynamics and control design space that provides good handling qualities for a given vehicle type and flying task. A new generation of spacecraft is now under development by NASA and commercial entities to replace the space shuttle and perhaps eventually return astronauts to the moon; these vehicles will have a manual control capability for several mission tasks, and the ease and precision with which pilots can execute these tasks will have an important effect on performance, mission risk, and training costs. No reference standards currently exist for handling qualities of piloted spacecraft.

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The simulation reported in this paper builds on the extensive work NASA's Apollo Program conducted on manual control during lunar landing, which had principal goals of establishing simulation requirements for both research and training, selecting response types

for attitude control (a response type being the mapping between the pilot's input and the commanded vehicle state, e.g., a movement of the pilot control stick to the right could command a constant roll rate to the right or a constant translational velocity to the right), and designing the control system for compatibility with other manual control factors [7–15]. Reference [16] presents an overview of the findings of these papers. The goal of the current study is to inform the design of any piloted vehicle that will land on any planetary body with significant gravity, with the initial design space being that of NASA's now-canceled Altair lunar lander. Though a specific lunar lander design is presented here, the results of this evaluation should apply broadly to any piloted vehicle with a similar task. A study conducted in May 2007 using an Apollo lunar module (LM) model found that a precision lunar landing could not be executed with tolerable workload using a rate command with attitude hold (RCAH) response type unless attitude guidance was present [17]. A subsequent simulation of a generic lander based on Altair, conducted in December 2008, studied much lower values of control power than were examined in the original Apollo studies as a function of rotation inceptor command sensitivity [16]. That study found that improved handling qualities could be achieved by careful selection of the sensitivity for a given control power, but the handling qualities were still not satisfactory for routine use with the RCAH response type and low control powers. The results of the current study, which should not be interpreted as an evaluation of Altair's handling qualities, may be used to determine from a system perspective how to most easily achieve satisfactory handling qualities, whether by introducing new response types, changing displays, or increasing control power.

The specific contributions of this paper lie in the evaluation of novel response types and display elements, including several guidance schemes, at a control power that is 15% of that used by the Apollo LM. The task is to conduct a precision landing with and without guidance, which may be required for future lunar outpost missions. Note that precision landing was not a requirement for the Apollo LM. A principal goal of the current simulation is to examine whether handling qualities are satisfactory with improved response types and displays without resorting to increasing the size and weight of the reaction control system (RCS) jets to increase control power. In this experiment, the vehicle design was updated with the mass properties, geometry, visual field of view, and propulsion system (both RCS and descent engine) specifications to match that of the Altair design in August 2009. The displays, inceptors, visual cues, control response types, and guidance algorithms were all designed from scratch because they were not available for Altair. There were three primary goals of the study: 1) determine which control response types and display features improved handling qualities; 2) evaluate whether the configurations allowed the landing task to be completed without guidance; 3) quantify the effects of the configurations on touchdown performance and propellant use.

The paper begins with a discussion of the experiment design, which includes details on the flying task, test matrix, and procedure.

A description of the vehicle design follows, where the lunar lander dynamics and control model is detailed along with four new control response designs. The motion-base simulator is described next, along with the cockpit instrumentation, guidance and displays, and piloting procedures. Results are then presented, and the paper concludes with a summary of key findings.

## Experiment Design

In November 2009 a piloted simulation assessed the handling qualities for a precision lunar landing task from terminal descent to touchdown. The primary experiment variables were the control response type and display feature set. These independent variables are expected to be the most crucial for determining handling qualities if the control power of the vehicle is held constant. While the control power must be selected early in the design process and is difficult to change because of mass implications, the response type and displays may be modified in software to compensate for deficiencies known to result from low thrust levels.

### Flying Task

The Altair vehicle was designed to carry four astronauts plus cargo from lunar orbit to the surface and return only the people to orbit in an ascent stage that is separable from the lower, descent portion of the lander. This experiment evaluated handling qualities for a precision landing task, from final approach through terminal descent to touchdown, following a trajectory based on the nominal Apollo trajectory [18]. Changes to the vehicle trajectory were made by firing opposing RCS jets to alter the attitude of the lander and correspondingly tilt the descent engine's thrust vector. Previously developed approach and landing task feedback guidance laws were used [17], and the corresponding velocity or attitude guidance cues were displayed to the pilot via cockpit instrumentation.

The task began at 500 ft altitude with a forward speed of 60 ft/s and a descent rate of 16 ft/s as shown in Fig. 1; for Apollo missions this was known as "low gate" and represented the point on the trajectory where the manual flying phase would begin [18]. At this point, the spacecraft was 1350 ft downrange from the designated touchdown point and pitched up 16 deg. The desired trajectory brought the spacecraft to a level attitude directly above the touchdown point, at an altitude of 150 ft with a descent rate of 3 ft/s. This rate of descent was held constant until one of the 6-ft probes attached to the lander legs made contact with the lunar surface. A shut-off command was then sent to the main engine, and the vehicle dropped until the legs settled on the lunar surface. For comparison, it also shows the uncontrolled trajectory that would result if no pilot inputs were made starting from an initial condition with vertical force equilibrium.

The dynamics of the lander's trajectory are confined to the vertical plane, with pitch attitude as the primary means of longitudinal

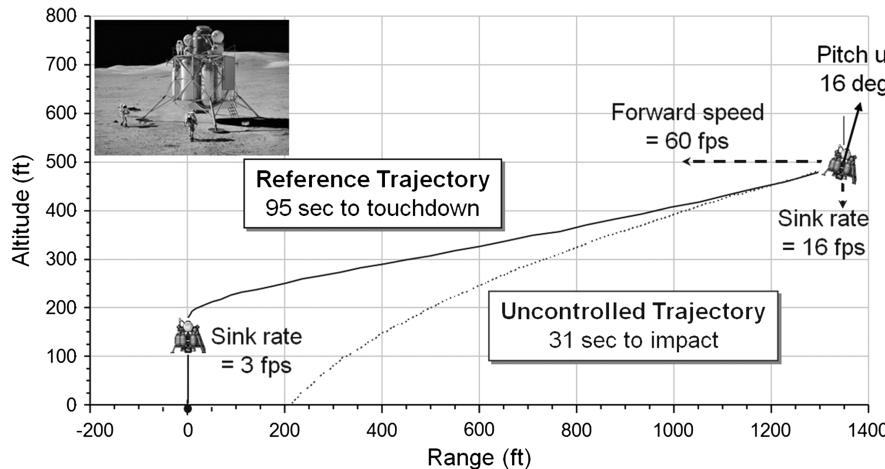


Fig. 1 Reference trajectory profile in the vertical plane.

trajectory control. To excite lateral dynamics, the initial condition included a lateral offset of 250 ft from the touchdown point so that the initial velocity vector did not point directly at the landing site. This lateral offset required the pilot to use roll attitude as a means of lateral trajectory control. The pilot similarly controlled pitch attitude to regulate the longitudinal trajectory, but the vertical trajectory was managed by the guidance and control system and the option of taking manual vertical control left to the pilot's discretion. Pilots activated manual vertical control only when they had difficulty centering the lander over the landing pad and needed extra time; in practice this happened very rarely for the configurations with good handling qualities and regularly for those with bad.

### Experiment Matrix

The experiment matrix was designed to answer the question of whether satisfactory handling qualities could be achieved with highly augmented control response types, better heads-down displays, or combinations of both. Two classes of control response types were developed, one class appropriate for the entire trajectory from low gate to touchdown but optimized for the “up-and-away” portion (altitude between 150 and 200 ft), and another class optimized for the final terminal descent portion of the trajectory from 150 ft to touchdown. This addresses the question of whether forcing the pilot to change his method of control halfway through the task creates more difficulty than the presumed benefit of having response types that are better tailored to the particular characteristics of the two portions of the trajectory.

The displays used in this simulation were adapted from aircraft glass cockpit and advanced rotorcraft designs [19]; the latter was used because of the perceived similarities between lunar landing and rotorcraft landings. These displays attempt to present vehicle state information in an intuitive way, and significant effort was devoted to scaling the display symbols' relative positioning to allow the pilot to fly the task without explicit guidance. Configurations were tested both with guidance, tailored to the particular up-and-away response type, and without guidance to ascertain the need for this extra feature as a function of control response type.

The order in which the pilots flew each configuration was carefully selected to minimize training effects. Each pilot flew the guidance cases before flying any nonguidance cases, and evaluated all of the configurations of one of the up-and-away response types before evaluating the configurations of the other up-and-away response type. The response type presented first was alternated for each pilot. Evaluations of the configurations for a given up-and-away response type were presented in random order to each pilot. Once all the guidance cases had been evaluated the pilots flew the three velocity increment (VINC) configurations without guidance, and with the time remaining flew as many RCAH configurations without guidance as possible. All 14 configurations of the experiment are shown in Table 1, however, not every pilot was able to complete all the nonguidance configurations, so those are considered part of the secondary matrix. All pilots completed each of the guidance configurations.

### Evaluation Pilots and Training

Eleven highly trained test pilots, comprising six Shuttle pilot astronauts, two Apollo LM pilots, and three NASA test pilots served as evaluation pilots. All pilots were male and had substantial training and experience as test pilots, logging an average of 7200 hours on various fixed or rotary wing and powered-lift aircraft. Relevant experience also included 13 Shuttle missions as pilot, ten as Shuttle commander, and two Apollo missions.

Pilots were allowed a two-hour training and familiarization session in the simulator cockpit, in which they practiced the flying task for each of the configurations with guidance and the VINC mode without guidance. Training for the nonguided RCAH response type was left until after data collection in the primary matrix was completed. The pilot was allowed to continue training until he judged his performance was consistent and he had developed appropriate piloting strategies. During data collection the pilots were given 10 min breaks each hour, and all pilots were able to complete configurations 1 to 4 and 9 to 14 (see Table 1) in a single day.

### Data Collection Procedures

The pilots were required to fly at least two data runs in each of the 14 configurations, with the option for as many additional runs as desired, and then provide Cooper–Harper handling qualities ratings [1], NASA Task Load Index ratings [20], and qualitative comments. The ratings were based on both the quantitative touchdown performance (see Table 2) and a qualitative evaluation of the manner in which the vehicle got to the end point. This overall assessment of desired, adequate, or inadequate performance was judged for the complete task of approach and landing. The values of desired and adequate performance bounds for key parameters were obtained from a survey of Apollo Lunar Module literature, the primary consideration being confidence that adequate performance would not result in the vehicle tipping over or damaging the legs; the 15 ft range error limit for desired performance is roughly comparable to the required landing accuracy for the automated system during normal conditions (10 ft).

A debrief session was held after all test configurations had been evaluated. The pilots were asked to fill out a one-page questionnaire designed to elicit high-level comments on cockpit displays, out-the-window displays, guidance cues, control response, and experiment design. The pilot's participation in the study concluded with a discussion of key insights and suggestions for future work.

### Lunar Lander Dynamics and Control Model

NASA's Altair vehicle [21] was in the early design stages when model development was completed in October 2009, so the model is based only on available propulsion system characteristics, mass properties, and geometry of the new vehicle. Some of the remaining aspects of the model, including the RCAH control system design and switching curve logic, approach trajectory, and touchdown performance requirements, were based on Apollo LM data from several sources [14,22,23], and the remaining aspects, including new control

**Table 1 Experiment configurations**

Configuration	Control response type (up-and-away)	Control response type (terminal descent)	Guidance type
1	Rate command with attitude hold	Rate command with attitude hold	Attitude
2	Rate command with attitude hold	Acceleration command	Attitude
3	Rate command with attitude hold	Translation rate command with position hold	Attitude
4	Rate command with attitude hold	Incremental position command	Attitude
5	Rate command with attitude hold	Rate command with attitude hold	None
6	Rate command with attitude hold	Acceleration command	None
7	Rate command with attitude hold	Translation rate command with position hold	None
8	Rate command with attitude hold	Incremental position command	None
9	Velocity increment command	Velocity increment command	Velocity
10	Velocity increment command	Translation rate command with position hold	Velocity
11	Velocity increment command	Incremental position command	Velocity
12	Velocity increment command	Velocity increment command	None
13	Velocity increment command	Translation rate command with position hold	None
14	Velocity increment command	Incremental position command	None

**Table 2** Upper limits of desired/adequate touchdown performance

Touchdown parameter	Desired	Adequate
Range to target point, ft	15	25
Horizontal velocity, ft/s	2	4
Descent rate, ft/s	6.5	8
DE propellant burn, lb <sub>m</sub>	35% remaining	10% remaining
RCS propellant burn, lb <sub>m</sub>	35% remaining	10% remaining
Roll angle, deg	3	6
Pitch angle, deg	3	6
Roll rate, deg	3	6
Pitch rate, deg/s	3	6
Yaw rate, deg/s	1	1.5

response types, displays, inceptors, and RCS prioritization logic were developed for this experiment. Figure 2 presents the coordinate frame and schematic of the Altair vehicle.

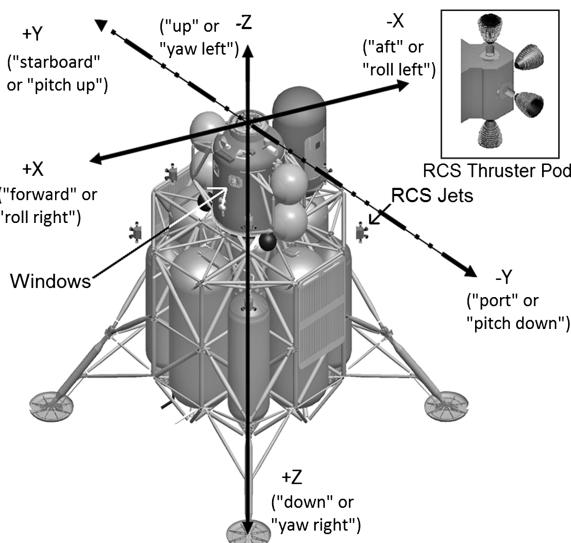
### Vehicle Mass Properties and Dimensions

The mass of the vehicle when the task begins is 55,900 lb<sub>m</sub>; it then varies due to consumption of propellant by the descent engine and RCS jets. During the final approach to touchdown phase the vehicle mass decreases by only 5% due to propellant consumption. Hence in this model it is assumed that moments of inertia are constant and that the vehicle center of mass (c.m.) location remains constant. Propellant slosh effects at the low angular accelerations available in this model were found to be negligible in a previous study [16] and were not modeled here.

The pilot's eyepoint is located about 30 ft above the ground when the vehicle has touched down, which corresponds to a little more than 9 ft above the lander's c.g. with the expected propellant load at low gate. The viewable angle below the +x axis from the commander's station is approximately 37 deg, which means that at low gate the view of the landing site is obstructed by the structure of the descent stage and the commander cannot see the intended touchdown point. In contrast, the Apollo LM had a downward viewing angle of about 65 deg, a design requirement specifically intended to help the commander determine the safety of the landing site during the final phase of lunar descent.

### Descent Engine

The descent engine is the spacecraft's main rocket engine, with a specific impulse of 449 s. For the landing task, its thrust force is used to regulate the descent rate and to apply coarse trajectory control in the horizontal plane by rolling and/or pitching the vehicle. In this model, the engine does not gimbal, and the thrust line passes through

**Fig. 2** Schematic of Altair Lander.

the vehicle c.m. Propellant mass budgeted for the piloted segment of the landing trajectory, including reserves, is 4774 lb<sub>m</sub>.

The descent engine thrust is directed along the negative body z-axis. During the flight phases from approach to touchdown, this thrust can be controlled by a throttle between 10 and 100% of the maximum value of 18,627 lb<sub>f</sub>. The thrust command,  $T_{\text{cmd}}$ , consists of two parts:  $T_{\text{cmd}}^o$  and  $\Delta T_{\text{cmd}} \cdot T_{\text{cmd}}^o$ .  $T_{\text{cmd}}^o$  is the equilibrium thrust whose vertical component of force balances the vehicle's lunar weight,  $mg_{\text{lunar}}$ , while compensating for vehicle roll ( $\phi$ ) and pitch ( $\theta$ ) angles:

$$T_{\text{cmd}}^o = \frac{mg_{\text{lunar}}}{\cos \phi \cos \theta} \quad (1)$$

The secondary part of the thrust command  $\Delta T_{\text{cmd}}$  is an increment derived from pilot input to the vehicle's rate of descent (ROD) command. In this command mode each inceptor "click" (movement of the inceptor out of detent and back into detent) increments the commanded descent rate by  $\pm 1$  ft/s; the descent rate is regulated within a deadband of  $\pm 0.1$  ft/s by a proportional feedback controller with a time constant  $\tau = 1.5$  s

$$\Delta T_{\text{cmd}}^{\text{ROD}} = \frac{m}{\cos \phi \cos \theta} \left( \frac{\dot{h}_{\text{cmd}} - \dot{h}}{\tau} \right) \quad (2)$$

Engine response to thrust commands is modeled as a first-order, unity-gain system with a time constant of 0.7 s. Hence the actual thrust produced by the descent engine  $T_{\text{DE}}$  lags the commanded thrust,  $T_{\text{cmd}} = T_{\text{cmd}}^o + \Delta T_{\text{cmd}}$ .

### Reaction Control System

There are four RCS pods mounted symmetrically on the corners of Altair's descent module, as shown in Fig. 2. The RCS jets cannot be throttled; their response to a command input begins with a pure delay of 10 ms followed by a first-order thrust response that reaches full thrust within an additional 13 ms. The RCS jet must then fire for a minimum of 40 ms, after which the thrust decays as a first-order response with time constant 110 ms. Propellant mass budgeted for the piloted segment of the landing trajectory, including reserves, is 444 lb<sub>m</sub>. RCS jets in this experiment are used exclusively for three-axis attitude control and not for direct translation. When the jets are fired in combinations of two or four there is no cross-coupling of thrust from one axis to another (for instance pitch moments coupling into roll), however, the nonzero moment of inertia term  $I_{xz}$  did cause coupling between roll and yaw.

### Inceptor Shaping

The state values commanded by the pilot were calculated by shaping the raw displacement of either the rotational hand controller (RHC) or translational hand controller (THC) using a linear-quadratic function, the same shaping function used in Apollo. This function is simply a weighted average of a linear function and a quadratic function:

$$\omega_{\text{cmd}} = \omega_{\text{cmd}}^* \left( \frac{(\delta - \delta_{\text{DB}}) + c_{\text{LQ}}(\delta - \delta_{\text{DB}})^2}{(1 - \delta_{\text{DB}}) + c_{\text{LQ}}(1 - \delta_{\text{DB}})^2} \right) \quad \text{if } \delta > \delta_{\text{DB}}; \\ \text{else } \omega_{\text{cmd}} = 0 \quad (3)$$

where  $c_{\text{LQ}}$  is the linear-quadratic shaping parameter,  $\delta_{\text{DB}}$  is the inceptor detent deadband, and  $\omega_{\text{cmd}}^*$  denotes the state value ( $p_{\text{cmd}}$ ,  $q_{\text{cmd}}$ , or  $r_{\text{cmd}}$ ) commanded at full throw of the inceptor ( $\delta = 1$ ). The parameter values were  $c_{\text{LQ}} = 1$ , the RHC deadband was 10% of full throw, the THC deadband was 30% of full throw, and  $\omega_{\text{cmd}}^*$  varied depending on the state variable being commanded by the particular inceptor.

### Attitude Control

Pilot attitude inputs are made with a three-axis RHC. This control inceptor is used for attitude stabilization and control along all three body axes; however, mode control is determined simultaneously for the roll and pitch axes (they are always in the same mode) and

independently for the yaw axis. Whereas roll and pitch commands may be generated from either of the below response types, the yaw response type is always RCAH.

#### Rate Command with Attitude Hold

This mode is in effect when the rate command in a particular axis is changing very quickly (in “response to urgent commands,” [14] which for Apollo was a change in rate command of 6 deg/s in 1 s) and the attitude rate error is outside a 0.3 deg/s deadband. Error signals in rate mode are generated as the difference between the actual and desired angular rates:

$$\begin{Bmatrix} p_{\text{err}} \\ q_{\text{err}} \\ r_{\text{err}} \end{Bmatrix} = \begin{Bmatrix} p - p_{\text{cmd}} \\ q - q_{\text{cmd}} \\ r - r_{\text{cmd}} \end{Bmatrix} \quad (4)$$

where  $p$ ,  $q$ ,  $r$ , are the roll, pitch, and yaw rates, respectively, along the vehicle body axes. The commanded rates are computed from the inceptor deflection and the inceptor shaping function described in the previous section.

When the attitude rate error drops below 0.3 deg/s and the RHC is not changing positions rapidly the control mode changes to attitude hold. The error signals for attitude hold mode are given by:

$$\begin{Bmatrix} p_{\text{err}} \\ q_{\text{err}} \\ r_{\text{err}} \end{Bmatrix} = \begin{Bmatrix} p \\ q \\ r \end{Bmatrix}$$

$$\begin{Bmatrix} \phi_{\text{err}} \\ \theta_{\text{err}} \\ \psi_{\text{err}} \end{Bmatrix} = \begin{bmatrix} 1 & 0 & -\sin \theta \\ 0 & \cos \phi & \sin \phi \cos \theta \\ 0 & -\sin \phi & \cos \phi \cos \theta \end{bmatrix} \begin{Bmatrix} \phi - \phi_{\text{cmd}} \\ \theta - \theta_{\text{cmd}} \\ \psi - \psi_{\text{cmd}} \end{Bmatrix} \quad (5)$$

where  $\phi$ ,  $\theta$ , and  $\psi$  are the current values of the vehicle Euler angles, and  $\phi_{\text{cmd}}$ ,  $\theta_{\text{cmd}}$ , and  $\psi_{\text{cmd}}$  are equal to the sum of the Euler angle values captured when attitude hold mode was last entered (for the particular axis) plus the integral of the commanded rates. That is:

$$\phi_{\text{cmd}} = \phi(t_0^{R/P}) + \int_{t_0^{R/P}}^t p_{\text{cmd}} dt \quad \theta_{\text{cmd}} = \theta(t_0^{R/P}) + \int_{t_0^{R/P}}^t q_{\text{cmd}} dt$$

$$\psi_{\text{cmd}} = \psi(t_0^Y) + \int_{t_0^Y}^t r_{\text{cmd}} dt \quad (6)$$

where  $\phi(t_0^{R/P})$ ,  $\theta(t_0^{R/P})$ , and  $\psi(t_0^Y)$  are the Euler roll, pitch, and yaw angles captured when those axes last entered attitude hold mode, and  $p_{\text{cmd}}$ ,  $q_{\text{cmd}}$ , and  $r_{\text{cmd}}$  are the time-dependent attitude rate commands from the RHC. Control moment commands are generated about the appropriate axes in accordance with the phase-plane relationship between error signals, as illustrated in Fig. 3 for the pitch axis [14]. In that figure,  $\alpha_p$  is the nominal angular acceleration achieved with two RCS jets (approximately 0.65 deg/s),  $\theta_{\text{DB}} = 0.3$  deg is the deadband for pitch attitude error,  $\theta_{\text{flat}} = 0.8$  deg is the magnitude of the flat boundary that enlarges the deadband region,  $q_{\text{HI}} = 1.4$  deg/s is the rate error at which the control system transitions from two-jet to four-jet control, and  $k = 0.01$  denotes the tradeoff between RCS jet propellant consumption and error settling time. The phase-plane relationships for the roll and yaw axes are identical except for a small difference in the angular acceleration,  $\alpha$ .

#### Acceleration Command

Acceleration command (ACEL) is related to attitude command through the inverse sine function, and so is essentially a rotational response type despite its label. For this reason the acceleration command response type was implemented on the RHC. It was only available to the pilot during the terminal descent portion of the trajectory (for given experiment configurations) because only that portion requires a nominally zero value of translational acceleration; the up-and-away trajectory requires a time-varying, nonzero value of acceleration so the pilot task in those circumstances would be to hold

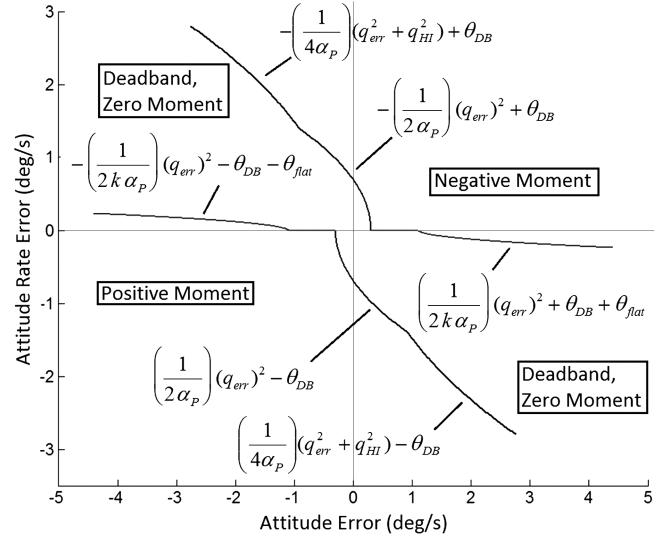


Fig. 3 Switching curves for RCAH.

the inceptor out of detent against a spring force, a situation that was unacceptable to the development pilots. During the terminal descent portion of the trajectory the pilot can make small corrections in velocity with pulse inputs to the RHC, a technique that was much more acceptable than a continuous displacement.

Acceleration command is implemented as a simplified version of the RCAH response type: only a single mode is available (attitude hold) and the raw value being output by the RHC is interpreted as a linear-quadratically shaped acceleration command, which is translated into roll and pitch commands via the following set of equations:

$$\phi_{\text{cmd}} = \sin^{-1}\left(\frac{m}{T_{\text{DE}}} \dot{v}_{\text{cmd}}\right) \quad \theta_{\text{cmd}} = -\sin^{-1}\left(\frac{m}{T_{\text{DE}} \cos \phi} \dot{u}_{\text{cmd}}\right) \quad (7)$$

where  $m$  is the current mass of the vehicle,  $T_{\text{DE}}$  is the current thrust of the descent engine, and the longitudinal and lateral acceleration commands are  $\dot{u}_{\text{cmd}}$  and  $\dot{v}_{\text{cmd}}$ , respectively. The maximum available acceleration was 2 ft/s<sup>2</sup>, which corresponds to about 20 deg of roll or pitch, and the commanded attitudes calculated from Eq. (7) were used with zero commanded attitude rates in the switching curve logic of Fig. 3. The difference between the RCAH and acceleration command switching curves are, in the case of the latter, the use of four jets whenever possible (not switching to two jets when the attitude rate error was small), no flat portion of the inner switching curves, and a time-propellant tradeoff value of  $k = 0.25$ .

#### Translation Control

Control of the vehicle’s translation rate was always done in the simulation through attitude commands rather than direct translation. The RCS jet thrust tested in this simulation was so low that firing the jets in parallel to achieve a pure translation force had little effect on translational velocity within the task time constraints. The following control response types commanded translational quantities (position and velocity) through manipulation of the pilot’s THC.

#### Velocity Increment Command

The velocity increment command (VINC) response type was available during both the up-and-away and terminal descent portions of the trajectory for the appropriate experiment configurations. Conceptually very simple, the VINC mode allowed the pilot to command a discrete change in the vehicle’s horizontal velocity with each discrete movement of the inceptor out of and back into detent. These discrete movements are referred to as “clicks” of the inceptor. Each click commanded a change in velocity in one axis of 10% of the total horizontal velocity; this scaling provided a straightforward way

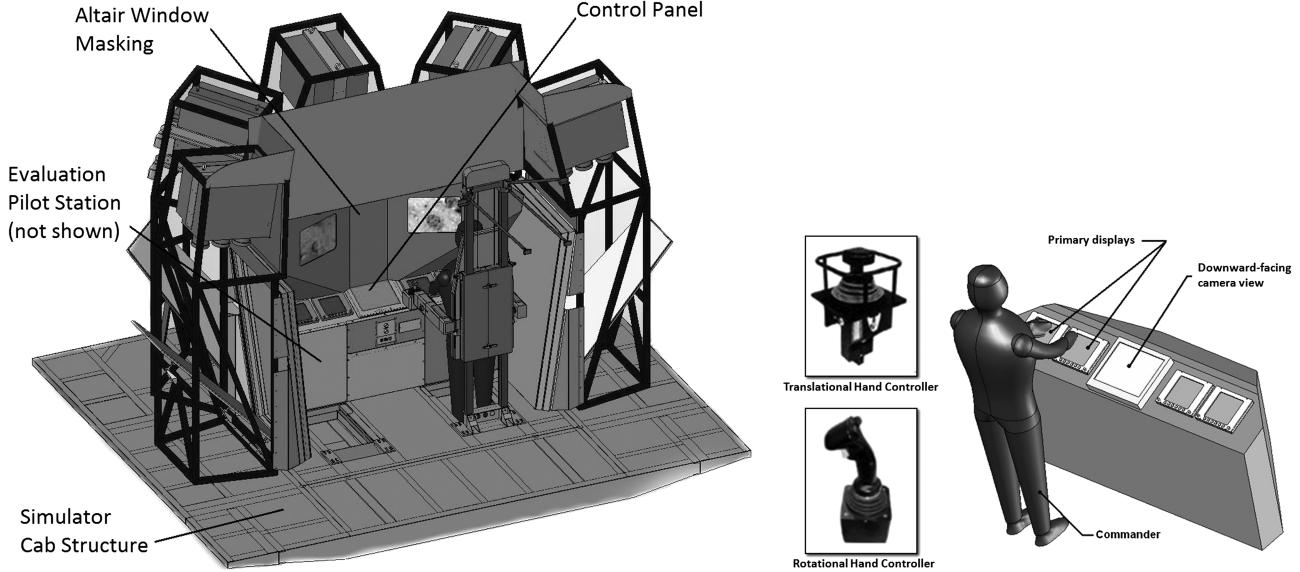


Fig. 4 Simulator cockpit layout.

to increase the fineness with which the pilot could change velocity as they got closer to the touchdown target and as their velocity was reduced. A minimum velocity increment of 1 ft/s was established as a lower limit for each “click.” This minimum increment did prevent the pilots from commanding precisely zero horizontal velocity, a feature to be remedied in future designs, but it did not distract pilots significantly because a command under 0.5 ft/s was achievable, which is significantly less than the desired 2 ft/s at touchdown (see Table 2). A blue ‘X’ was displayed on the pilot’s horizontal situation display indicating the current value of the velocity command.

VINC was implemented on the THC and the velocity changes were aligned with the inceptor axes. The roll and pitch commands to the control system were calculated using Eq. (7), and the acceleration commands were related to the velocity errors in each axis through the following transfer function:

$$\frac{a_{cmd}}{v_{err}} = 14 \frac{s + 1.5}{s + 150} \quad (8)$$

The attitude errors were then calculated from Eq. (5), the attitude rate errors were simply the attitude rates themselves because the target rate was zero, and the RCS firings were controlled by the switching curve logic expressed in Fig. 3 subject to the same caveats given for the acceleration command system. The maximum roll or pitch angle that the control system would command, regardless of the velocity error, was 30 deg per axis.

#### *Translation Rate Command with Position Hold*

The translation rate command with position hold (TRCPH) response type is a highly augmented control mode that was adapted from advanced rotorcraft control systems. It is the translational analog to the RCAA response type for attitude: when the THC is out of detent the pilot is commanding a translational velocity proportional to inceptor displacement, and when it is back in detent the control system automatically holds a lateral position. The maximum velocity that could be commanded was 5 ft/s at full throw of the THC, and the raw inceptor output was shaped with a linear-quadratic filter. This response type was only available for the terminal descent portion of the trajectory in certain experiment configurations because of its limited velocity range. The control mode would change from translation rate command to position hold whenever the THC was in detent (commanding zero rate) and the total horizontal velocity was under 1.5 ft/s. No explicit indication of the commanded translational velocity was shown to the pilots on their heads-down displays.

The implementation of TRCPH is identical to that of VINC mode with the exception of the shaping filter. Whereas the VINC mode

commanded a cumulative velocity based on all previous inputs, TRCPH commanded a single velocity based only on the current displacement of the THC. The transfer function used to calculate an acceleration from a given velocity error is shown in Eq. (8), the attitudes required for given acceleration commands are given by Eq. (7), and the same saturation limits of 30 deg per axis apply. For the position hold feature of this response type a separate, second-order transfer function is used to calculate the acceleration command for a particular position error; otherwise the rest of the logic remains the same. The transfer function from position error to acceleration command in each axis is

$$\frac{a_{cmd}}{x_{err}} = 6 \frac{s + 0.2}{s^2 + 4s + 20} \quad (9)$$

where  $x_{err}$  is the position error in a particular horizontal direction relative to the vehicle’s yaw angle. Both submodes of TRCPH used Eqs. (5) and (7) to calculate attitude errors for the switching curves with zero commanded attitude rate.

#### *Incremental Position Command*

The increment position command (IPC) response type was the most highly automated control system tested in this experiment, and it can be likened to a “target designator” system in which the pilot simply designates a landing point and the vehicle determines how to roll and pitch to get to that point. This mode is only available in certain configurations for the terminal descent portion of the trajectory, and when the pilot engages the response type the control system first enters a “velocity-nulling” mode. This intermediate mode, during which all pilot inputs are inhibited except for commands to change response types, is active as long as the translational velocity is above 1.5 ft/s. At this time the system is essentially in a translation rate command response type with a velocity command of zero. When the velocity drops below 1.5 ft/s and the control system begins to actively hold position, a blue box appears on the pilot’s horizontal situation display indicating the currently-held position. Each time the pilot moves the THC out of detent and back into detent (one click as discussed in the VINC section) the held position is incremented 5 ft in the appropriate direction. In an early implementation of this response type the blue box indicating held position was not available, so the first four pilots evaluated this response type without the aid of the touchdown point designator. The results for those two configurations are reported separately in the Results section.

This response type is implemented using the transfer function of Eq. (9) to calculate an acceleration command, saturated at  $\pm 2 \text{ ft/s}^2$ ,

for a given position error in a particular axis. That acceleration command is converted to an attitude command using Eq. (7), which is then converted to RCS firing sequences using the switching curve logic discussed for the acceleration command response type with zero commanded attitude rate.

### Guidance Laws

The velocity and attitude guidance laws used in this experiment were adopted without modification from an earlier lunar lander simulation; the detailed derivation may be found in [17]. These laws were designed to guide the pilot along a reference trajectory (see Fig. 1) from final approach through terminal descent to lunar touchdown.

Attitude guidance cues were presented to the pilots as errors from the desired vehicle states. These errors were computed as the differences between the guidance and actual values of the roll and pitch angles. The velocity guidance cues were presented both as errors (on the pilot's primary flight display) and as absolute quantities (on the horizontal situation display).

### Simulation Environment

The experiment was conducted in the vertical motion simulator (VMS) at NASA Ames Research Center. The VMS is a large motion-base simulator [24] that has been used for numerous handling qualities evaluations [25]. Six degree-of-freedom simulator motion was used for the experiment because the reference trajectory was dynamic, featuring translational accelerations up to  $0.35 \text{ ft/s}^2$  and roll/pitch angular accelerations up to  $1.3 \text{ deg/s}^2$ ; the VMS cab also provides excellent visual and auditory cues.

The simulator provided a similar cockpit configuration to that used in Apollo, as illustrated in Fig. 4. The evaluation pilot occupied the left station; the right station was occupied by the experimenter during training runs but not during data collection runs. Each pilot station provided a three-axis RHC and a three-axis THC mounted on the right and left armrest, respectively. A button on top of the RHC toggled between manual and automatic control of the vehicle's rate of descent. Up/down motion of the THC adjusted the commanded value of the rate of descent as long as manual vertical mode had been engaged.

A simulated view of the lunar landscape was projected on a set of five noncollimating flat screen rear projection color displays. The display had a large field of view: 77 deg vertical and 225 deg horizontal, and were approximately 3 ft from the pilot's eyepoint. Masking was applied in front of the out-the-window displays to restrict the usable field of view to be consistent with the 11" by 13" windows on Altair. This restriction is a departure from previous lunar lander experiments in the VMS, however, the precision landing task was essentially head-down, and the pilot's attention was focused

primarily on the cockpit instrumentation rather than the view outside the cockpit.

### Cockpit Instrumentation

Cockpit displays were mounted on a console and comprise two nine-inch flat-panel monitors at each pilot station and a 15-in. flat-panel monitor in the center. The pilot station displays are shown in Figs. 5a and 5b; the center monitor showed the view that would be captured by a camera positioned five feet in front of the descent engine nozzle (along the  $+x$  axis) and oriented along the body  $+z$  direction (downward-facing). That view showed only what would be captured by the camera and did not contain symbology; pilots found it useful for situational awareness when positioned over the landing pad, but not during the approach to the pad.

The screen on the pilot's left (Fig. 5a) is referred to as a horizontal situation display (HSD); many aspects of this display were adapted from advanced rotorcraft displays that have been shown to improve handling qualities for landings without a clear view of the desired touchdown point [19]. The white dot in the center of the HSD represents the spacecraft (ownship), and the pentagon in the shape of a "home plate" or "dog house" represents the landing site. The line pointing forward from the ownship represents the vehicle's horizontal velocity vector, and the open circle is the vehicle's current horizontal acceleration. That acceleration cue is referenced to the tip of the velocity vector so the line always grows or shrinks in the direction of the circle. The current yaw angle is represented by a white arc near the top of the HSD; that arc also provides a scale for the range and velocity indicators (currently 1500 ft and 120 ft/s in Fig. 5a). To avoid symbol clutter when close to touchdown, the HSD undergoes two discrete scaling steps at velocities of 30 ft/s and 10 ft/s. The current range to target in feet is next to an  $R$  on the left side of the HSD, while altitude information is to the right. The large 421 in Fig. 5a represents the current altitude in feet above ground level. The bar just to the left of the altitude readout represents the altitude rate, with each of the white tick marks representing five ft/s. A digital readout of the altitude rate is shown below this bar, currently reading "-15" ft/s. The white diamond superimposed on top of the altitude rate bar represents the altitude acceleration and is referenced to the tip of the rate bar; in this example the diamond is above the tip of the tape, which shows that the tape will grow toward the diamond and therefore the descent rate will be reduced. The open box on the tape is a reference for the pilot of the target descent rate at touchdown, 3 ft/s. The throttle setting of the descent engine is represented by a tape on the far right of the screen, and the propellant remaining (referenced to 100% at the start of the run) is below the throttle, reading "PROP 98%". Mode annunciators at the top of the HSD tell the pilot what the up-and-away response type is (left), what the terminal descent response type is (center), and whether the vertical trajectory is being manually or automatically controlled (right).

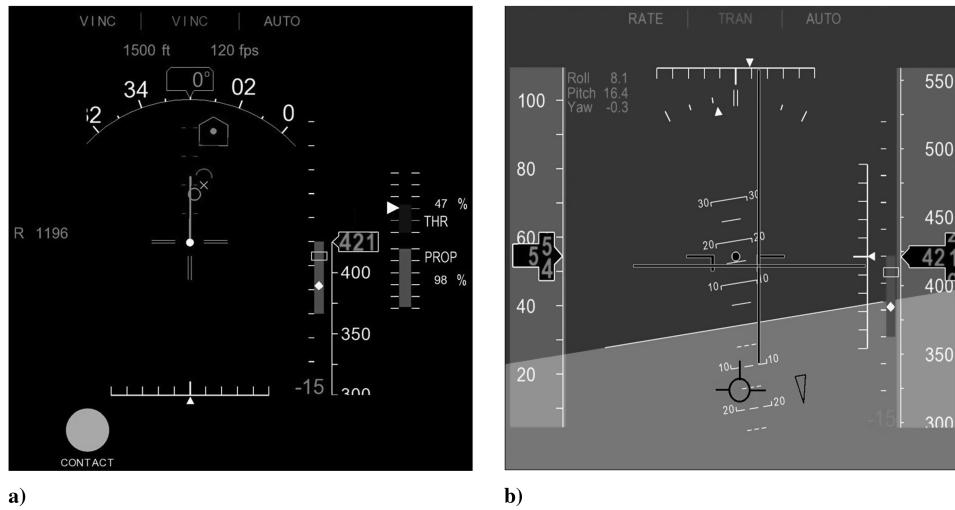


Fig. 5 Pilot station heads-down displays: a) horizontal situation display, b) primary flight display.

At the bottom of the screen are a yaw rate indicator and a touchdown indicator light. When velocity guidance is present a semicircle on the HSD represents the guidance velocity, so the pilot's job is to place the tip of the velocity vector in the center of the semicircle. The  $x$  represents the commanded velocity when in velocity increment mode.

#### Piloting Technique

The following procedures were reviewed with each pilot during the initial briefing. The pilots practiced these procedures during the training and familiarization simulator session before data collection.

#### Vertical Speed Control

Vertical trajectory control is accomplished by manual or automatic commands to the descent engine throttle. The simulation task began in automatic vertical control mode, which reduced the descent rate from an initial value of 16 ft/s at 500 ft altitude to 3 ft/s at approximately 150 ft altitude in the vicinity of the landing site and maintained that rate to touchdown. Pilots were allowed to switch to manual mode (by pressing a button on top of the RHC) if they needed extra time to establish the vehicle in a stable descent above the landing site. No vertical velocity guidance was present.

#### Horizontal Speed Control

Horizontal control of the lander was always done using attitude commands, whether explicitly using the RHC or implicitly through the control system using the THC. The pilots switched to the terminal descent response type when they were within 50 ft horizontally of the touchdown point and below a translational velocity of 5 ft/s; this response type change was accomplished by sliding a switch on top of the RHC. During data collection pilots were free to make their own decision as to when to make the control mode change, but they were required to make a switchover at some point during any run that included two response types.

The piloting task was always heads-down using the PFD and HSD, and the precise piloting requirements depended on the configuration of response types and guidance. When attitude guidance was present the pilot was required to center flight director needles on the PFD using the RHC, frequently using a pulse technique to make precise inputs. When velocity guidance was present the pilot would use the THC to command a particular horizontal velocity (represented on the HSD) to lead the velocity vector and maintain it within the semicircular velocity guidance symbol. When no guidance was

present in the velocity command modes the pilot would try to slow down according to a mental schedule of ranges and velocities, frequently using the relative positions of the touchdown symbol and velocity vector as a guide.

## Results

Data collection was conducted with 11 pilots between 26 October and 13 November 2009. The subjective handling qualities ratings given by the pilots included Cooper-Harper ratings and NASA Task Load Index (TLX) ratings. Objective data were captured during the simulation on the touchdown range dispersions and RCS propellant usage. Because of the large number of experiment configurations tested only select results will be presented. The response type abbreviations used in this section are RCAH, VINC, ACEL, TRCPH, incremental position command without predicted touchdown point (IPC), incremental position command with the predicted touchdown point (IPC-TD), attitude guidance (attitude), velocity guidance (velocity), and no guidance (state).

#### Cooper-Harper Handling Qualities Ratings

Pilots used the Cooper-Harper scale to assign handling qualities ratings from 1 (best) to 10 (worst) based on their assessment of task performance and effort. It is an ordinal scale, which means, for example, that the difference between Cooper-Harper ratings (CHR) of 1 and 2 is not the same as the difference between ratings of 3 and 4. Ratings of 1, 2, and 3 on the Cooper-Harper scale correspond to Level 1 handling qualities, which are a general requirement for normal operations of flight vehicles. Ratings of 4, 5, and 6 correspond to Level 2, which may be acceptable for some off-nominal conditions, and ratings of 7, 8, and 9 correspond to Level 3, which is acceptable only for transition to a safe mode after a major failure or disturbance. Desired performance is necessary (but not sufficient) for Level 1 ratings, and adequate performance is necessary (but not sufficient) for Level 2 ratings. A rating of 10 indicates that control would be lost in that configuration, and is represented here as "Level 3+".

The CHR data for all the cases in which guidance was available is presented in Fig. 6, and groups in that figure correspond to the same terminal descent response types. The shades of gray represent handling qualities Levels 1, 2, and 3; black represents a CHR 10. The  $x$ -axis labels in that figure refer to the up-and-away response type, the terminal descent response type, and the type of guidance available to the pilot.

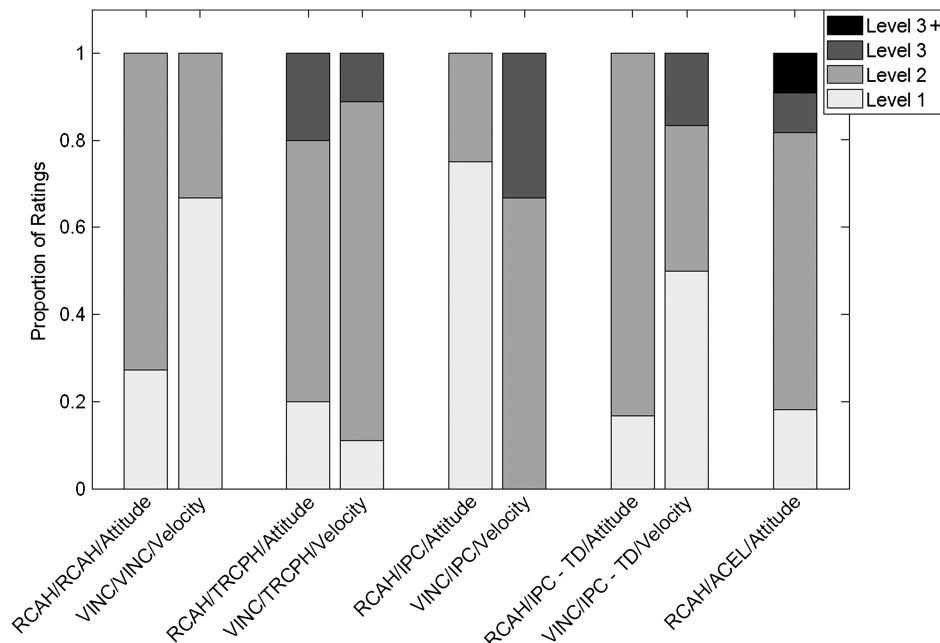


Fig. 6 Cooper-Harper ratings for all configurations with guidance present.

Comparing the first two bars of Fig. 6 shows that the configuration in which the pilot flew the VINC response type all the way to landing using velocity guidance has substantially improved handling qualities relative to the baseline RCAH response type flown with attitude guidance. Two-thirds of the pilots gave the pure VINC mode a Level 1 rating, while only three out of 11 rated RCAH Level 1. The median ratings for those two configurations are relatively close but still in different Levels, with CHRs of 3 and 4, respectively. Interestingly, the worst rating for the pure VINC mode was a CHR 6, while the worst rating for pure RCAH was a 4. Two of the pilots who gave the pure VINC mode a Level 2 rating objected to the lack of direct control over the lander's attitude; they felt that commanding a higher-order state like velocity and allowing the automatic system to control attitude was giving up too much control over the flight path and trajectory of the vehicle even though they found direct control of attitude to be a more technically demanding task than control of velocity. Research with pilots more familiar with each response type is needed to determine whether this objection would persist or disappear with additional training.

Another important comparison that may be made in Fig. 6 is the relative utility of a second response type designed for the terminal descent portion of the trajectory: additional mental demand is associated with changing piloting technique halfway through the task. With the exception of the configuration RCAH/IPC/Attitude, handling qualities degrade every time the pilot is required to change response types for the second half of the task. Pilots explained this degradation as being the result of the additional mental workload required to remember the satisfactory conditions for switching response types, changing mental models, and in some cases inceptors, for their interaction with the vehicle, making the physical switchover of the response type with the RHC slider switch, and the transient behavior of the vehicle as the new control mode became active or as the pilot changed their focus on the displays. It is possible that additional training would remove some of the mental effort required to change response types, or that giving the pilot more flexibility in deciding when to switch response types would improve the handling qualities ratings in comparison with the pure response types, but at the level of training present for this experiment it was not useful to have a second response type available.

Pilots generally prefer to be able to accomplish a task without relying solely on guidance, and previous studies showed that the precision landing task with RCAH requires either guidance or a large amount of training with high control powers. The VINC response type, in contrast, was well suited to this task with or without guidance

as shown in Fig. 7. This figure shows only the configurations in which VINC was the up-and-away response type, and groups the same terminal descent response types together with and without guidance. Differences in handling qualities ratings are minor between the guided and unguided cases and pilot comments confirmed that velocity guidance was only really useful as a training aid; once the pilot had flown about a dozen runs with VINC they tended to rely more on state data than guidance and used piloting techniques they developed on their own.

Figure 7 shows that the most useful of the terminal descent response types for VINC mode was IPC-TD. While pilots did not like being required to switch response types at fixed values of range and velocity, they did report a preference for having the option to change to IPC-TD once they had established the vehicle in a stable terminal descent. This was preferred because the IPC-TD and IPC systems tended to exhibit excessive overshoot when position commands exceeded 25 ft, and because once position hold was engaged there was little for the pilots to do. They reported that the only configurations in which they were able to look out the window and devote time to acquiring reasonable situational awareness about their trajectory were the configurations with IPC or IPC-TD.

The pilot ratings and comments suggest that the VINC mode should be optimized and examined in more detail as a way to improve handling qualities without increasing the size of the RCS thrusters. This is the only alternative method found to date that would provide overall Level 1 handling qualities (defined as greater than 60% of pilots giving a CHR of 1–3), and it is the only known method that can achieve this without the use of explicit guidance cues. The IPC-TD system should also be investigated as an optional feature for pilots to use when they feel comfortable turning position control over to the automatic system. More sophisticated control systems can reduce the overshoot for large position increments, and allowing pilots the freedom to change response types at any time would provide them the freedom to employ their preferred flying techniques.

#### NASA Task Load Index Ratings

The utility of the NASA TLX scale lies in the quantification of workload in six different categories, which allows the experimenter to determine what type of workload is contributing to the difficulty of the task and not simply how much total workload is involved. This distinguishes the TLX scale from the Cooper–Harper scale because performance and workload are not jointly considered in the selection of a single rating, but are explicitly rated independently. The figures

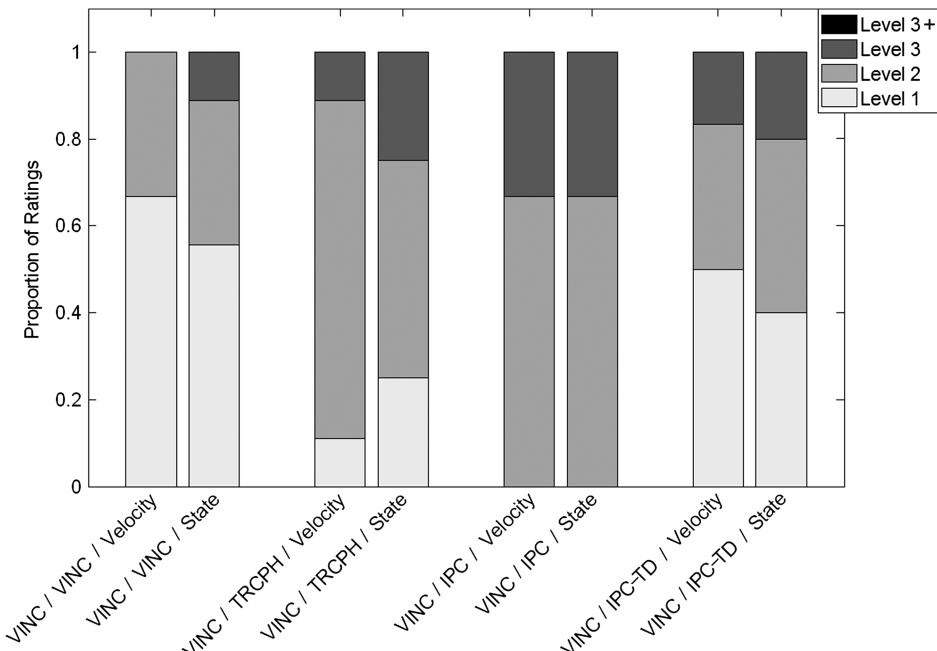


Fig. 7 Cooper–Harper ratings for velocity increment mode with and without guidance.

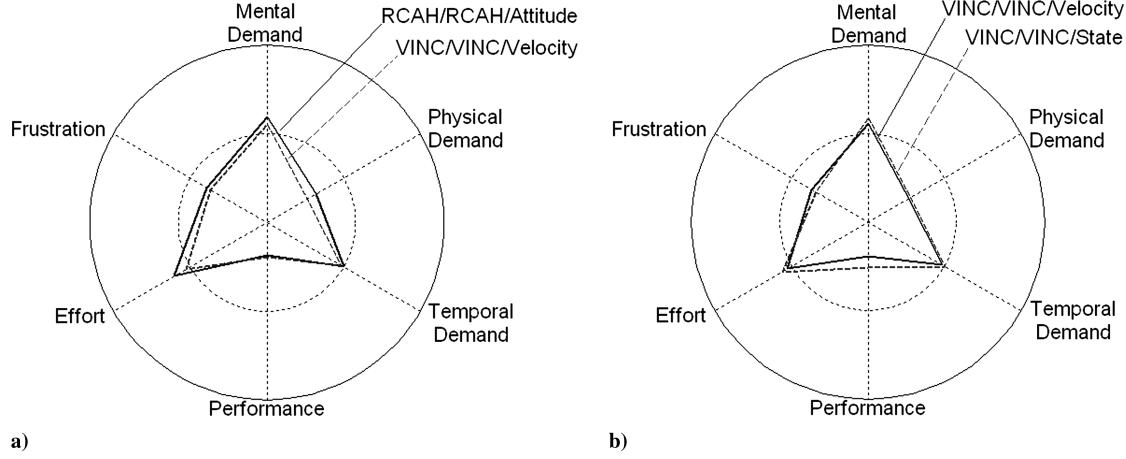


Fig. 8 TLX ratings for selected configurations: a) RCAH vs VINC with guidance and b) guidance comparison for VINC.

in this section plot the six TLX components as a function of radial distance from the center: the maximum (worst) rating is plotted at the outer, solid line, an intermediate rating is at the inner dashed line, and zero workload in a particular category (best) would be plotted at the center of the circle. The performance rating given by the pilots was opposite in sense to the other five categories (100 being good, 0 being bad), so in these results the original performance rating is subtracted from 100 and all categories have the same sense.

The TLX ratings for the pure up-and-away response types are shown in Fig. 8a, and the comparison of guided and unguided pure VINC modes in Fig. 8b. In parallel with the CHR results (see Figs. 6 and 7), the VINC mode receives better ratings than the pure RCAH mode when guidance is present. Temporal demand and performance are rated about the same between the two configurations, but the other four elements of workload are all slightly better with the pure VINC mode. These differences are not statistically significant at the 5% level. Also paralleling the CHR results are the similarities between the guided and unguided pure VINC modes; without guidance the workload categories are rated slightly worse than the guidance cases, but the differences are very small. The main conclusions discussed in the Cooper-Harper section are confirmed by the TLX ratings: pure VINC mode is rated more favorably than pure RCAH mode, and guidance does not provide significant improvements in workload for the pure VINC mode.

The TLX ratings provide new insights in comparison with CHRs when the use of a terminal descent response type is considered. Figure 9 shows the six components of TLX for the RCAH response type (Fig. 9a) and VINC response type (Fig. 9b) for each of the terminal descent types with guidance available. While the pure modes do well in comparison with the mixed modes, the conclusion from the CHRs that they are the best modes is not supported by the

TLX data. In the case of RCAH the IPC-TD response type is the clear favorite in every category except performance and temporal demand, with a statistically significant difference in overall ratings (an analysis of variance gives  $p = 0.032$ ). For the VINC mode the IPC-TD response type is rated equivalently to the pure mode, showing that the benefit of the IPC-TD response type roughly outweighs the additional workload required to switch to that mode for terminal descent. The reasons for this were given in the Cooper-Harper ratings section and include the principal fact that in most cases with this terminal descent response type the pilot was able to spend the last tens of seconds of the simulation run looking out the window and gaining situational awareness about the vehicle, trajectory and landing site. In no other response type was this possible. The lower performance ratings received by the IPC-TD response type were due to the already-mentioned tendency for the control system to overshoot in position when large position increments were commanded. These results indicate that position command response types should receive further examination.

#### Pilot-Vehicle Performance

The handling qualities ratings indicate the overall level of compensation and performance a pilot experienced in the execution of a particular task. It is necessary to examine the objective performance as determined by numerical performance metrics to help determine whether it was the compensation or the performance itself that was the principal factor in the handling qualities rating of a particular configuration. The most important performance parameter for this precision landing task, given that very few of the touchdowns would have resulted in the vehicle tipping over, was the range to the target at touchdown. The most important parameter from a vehicle

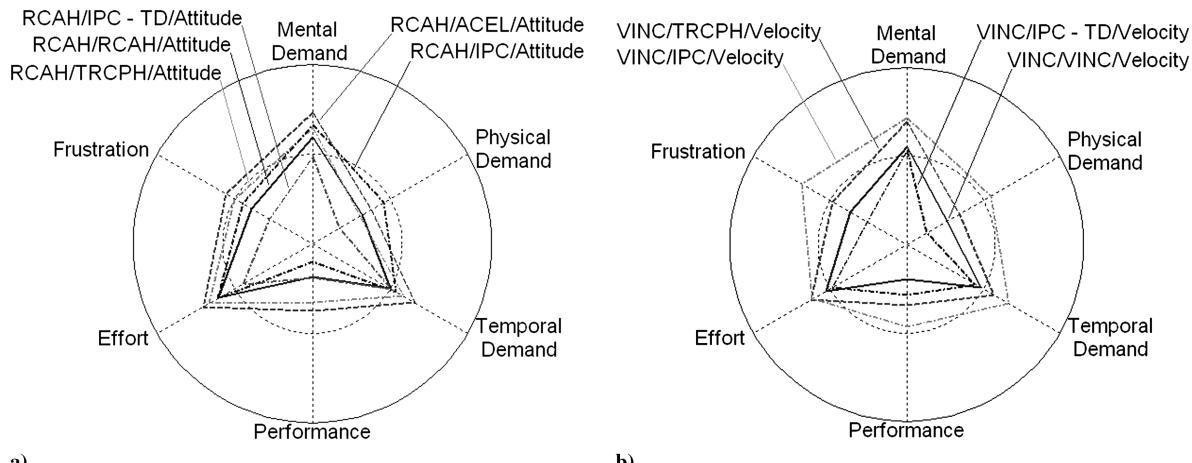


Fig. 9 TLX ratings for guidance configurations.

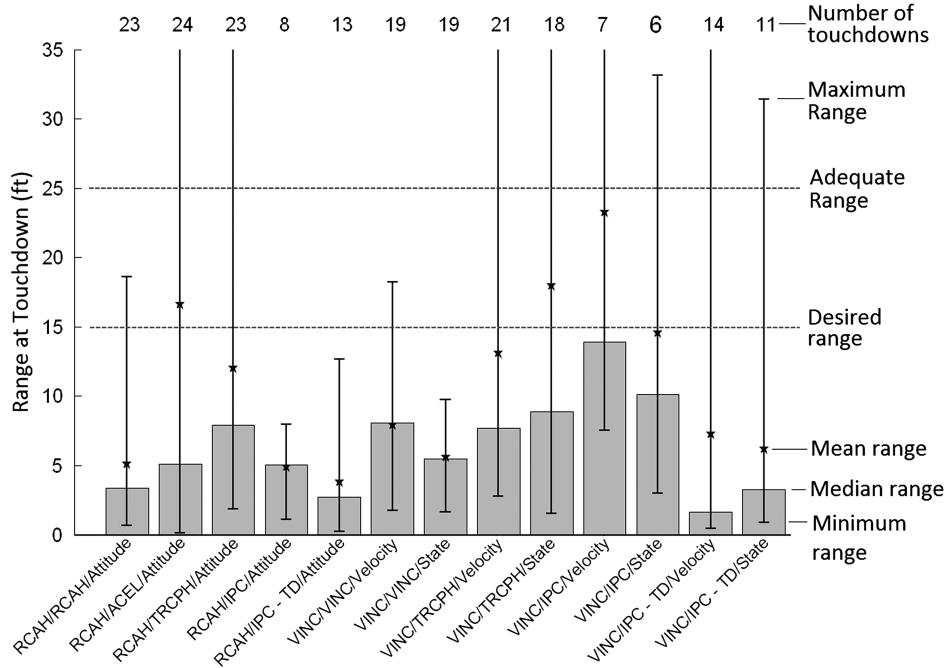


Fig. 10 Touchdown range dispersions.

design perspective is the amount of propellant required to execute the task.

The median, mean and absolute ranges of all the pilots' touchdown ranges are shown in Fig. 10. Here, touchdown range is defined as the horizontal distance between the centerline of the vehicle after it came to rest on the lunar surface and the center of the landing site; the desired range was 15 ft, adequate was 25 ft. The maximum range for many of the configurations is not shown because the pilot landed far from the touchdown site, well outside the adequate range and often because he had given up on achieving good touchdown performance for that particular run. The presence of these large range errors emphasizes the difficulty of the precision landing task and ever-present possibility of inadequate performance even after a string of successful runs. Figure 10 also demonstrates the utility of having a position command response type (IPC) and, more specifically, one that shows the actual designated touchdown point (IPC-TD): while

the handling qualities ratings were generally best for pure response types, the use of IPC-TD improved the median touchdown accuracy over the pure types. Finally, the pilots commented that following explicit attitude guidance with RCAH required significant attention and so received relatively poorer handling qualities ratings than the VINC modes, but that additional attention appears to have resulted in more accurate touchdowns.

The amount of RCS propellant consumed by the principal experiment configurations is shown in Fig. 11. That figure shows there is little difference in RCS propellant consumption across all these experiment configurations, a conclusion that is reasonable because the amount of consumed propellant is dependent on changes in the vehicle's attitude during the reference trajectory, and each configuration resulted in approximately the same set of roll and pitch maneuvers. This result is useful to a lunar lander designer because it suggests that the control response type may be selected without

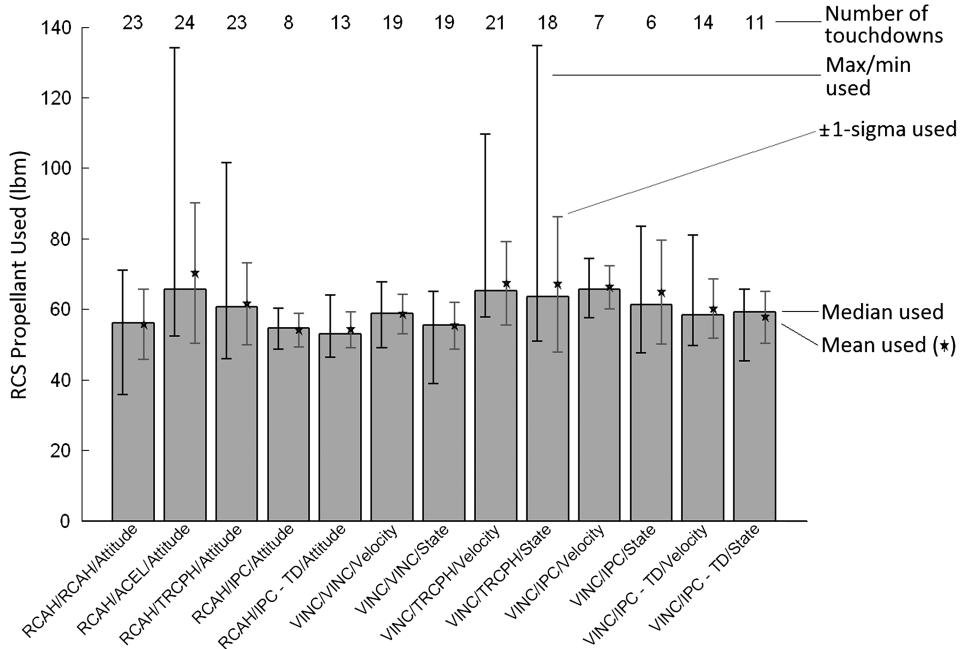


Fig. 11 RCS propellant consumption.

having to worry about its effect on propellant mass. The descent engine propellant consumption showed even less variability among the configurations. The only circumstance in which additional descent engine propellant was consumed was when a large position error during terminal descent drove the pilot to reduce his descent rate and therefore spend additional time and propellant executing the task.

## Conclusions

Eleven NASA astronauts and test pilots participated in a simulation evaluation of the handling qualities of a lunar lander during the final phase of a precision approach and landing task using a variety of control response types and guidance and display features. The control response types included the baseline RCAH type used by the Apollo LM, a more highly augmented velocity increment response type, and three response types designed for use only during the terminal descent portion of the trajectory. The displays included features adapted from aircraft glass cockpit and advanced rotorcraft designs, and configurations were presented to the pilot that either did or did not contain guidance. The principal goal of the experiment was to determine whether Level 1 handling qualities could be achieved with low control power for some combination of these control response types and display features, or whether increasing the vehicle's attitude control authority was the only way to reach Level 1. Secondary goals were to find response types that could be flown without guidance, and to measure touchdown range and propellant consumption as a function of the experiment variables.

The new velocity increment command response type received a majority of Level 1 ratings, meeting the criteria for overall Level 1 handling qualities. This represents a substantial improvement over RCAH. Whereas the current simulation and several previous experiments have consistently shown that this latter response type, used by the Apollo LM but with higher control authority, is given Level 2 ratings, only the velocity increment command response type tested here was Level 1. In addition, this response type received nearly identical ratings whether guidance was present or not; in contrast, the baseline rate command system proved extremely difficult to fly without guidance given the limited training time available to each participant.

The response types designed for use in the terminal descent portion of the trajectory did not improve handling qualities, with one possible exception, and generally showed satisfactory performance. While these response types were optimized for the small translation maneuvers required in the final descent, the additional mental effort required to change flying techniques, monitor different instruments or change inceptors outweighed most of the benefits of these additional control modes. The incremental position command response type afforded the pilot time to increase his situational awareness during terminal descent and improved touchdown accuracy. This response type could be a beneficial feature for pilots to engage at their discretion.

Though this experiment used a specific lunar lander design, Altair, the results of this evaluation should be broadly applicable to any powered vehicle in which direct thrust (rather than aerodynamic lift) is used to support the weight and attitude control is used for horizontal maneuvering. The results are also applicable when a pilot's control, in this case attitude acceleration, is four time derivatives away from the variable to be controlled, the lunar lander's horizontal position.

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