

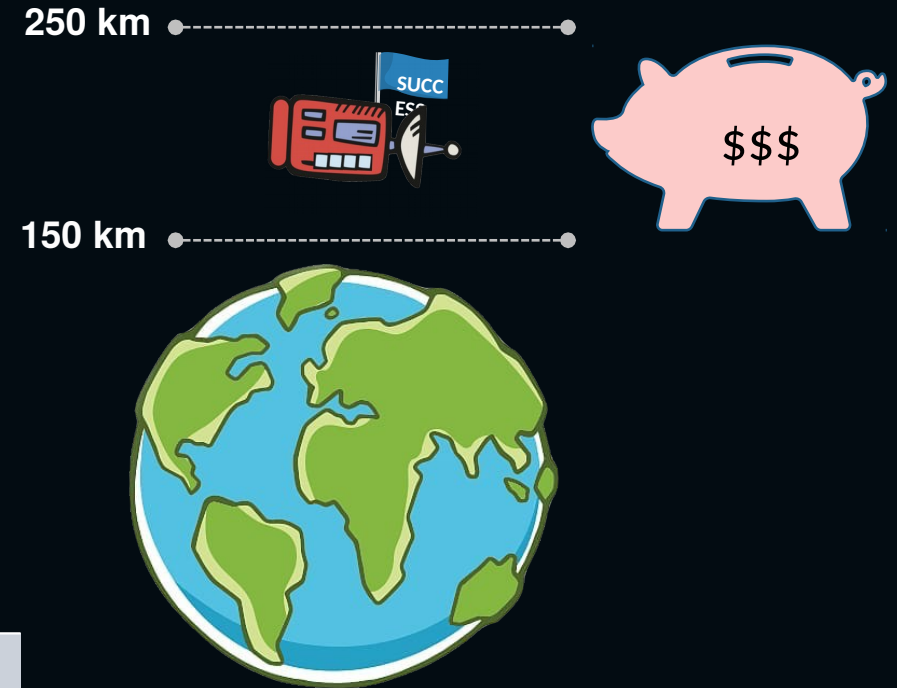
BREATHING SATTELITE

PARTICLE COLLECTION FOR AIR ENGINE IN SATELLITE

Team #38

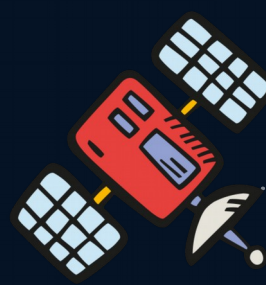
Very Low Earth Orbit (VLEO) — most commercially viable orbit

$Mass \sim Height^3$



Physical Parameters	Model Predictions		
Orbital altitude (km)	200	400	800
Satellite Mass (kg)	24,4	194,8	1558,6



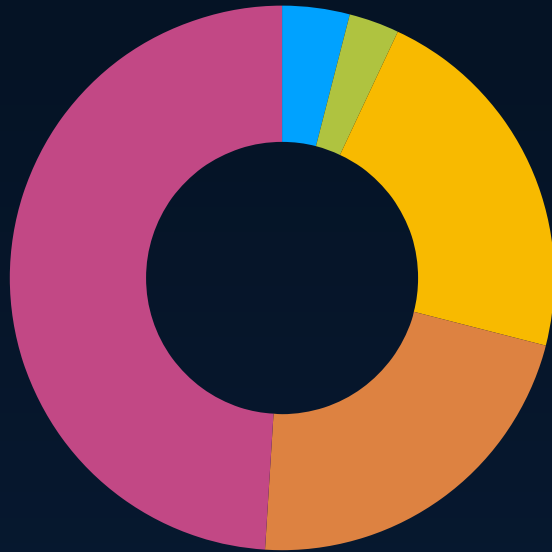


PROBLEM STATEMENT

- Short Lifespan of Satellites
in Very Low Earth Orbit
(150 - 250 km)
because of fuel limitation



Forecasted Number of satellites in 2017-2026



■ Smallsats (<500 kg)

OUR SOLUTION:

ACTIVE PARTICLE COLLECTOR FOR SATELLITE ENGINE



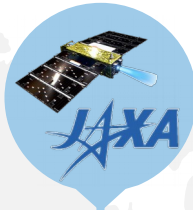
- **Active system**
- collection of air particles
- thrust generation to overcome the drag

COMPETITORS

There is no ready solution with use
atmospheric air as fuel



European Space Agency Project



Japan Aerospace Exploration
Agency

The Super Low Altitude Test
Satellite "TSUBAME"

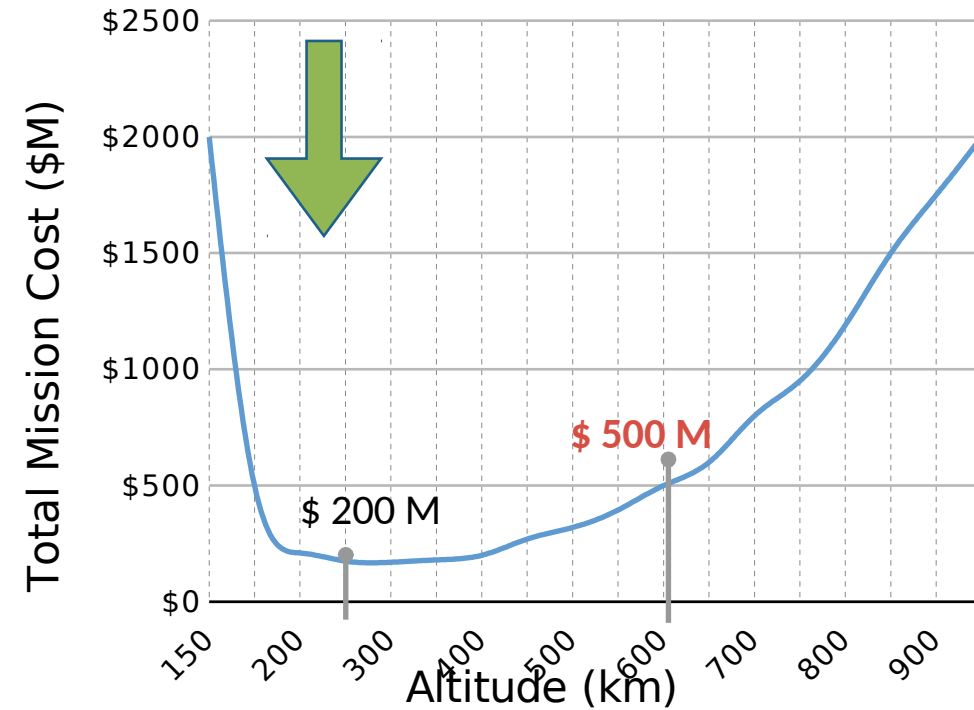
Design life	~ 2 years
Altitude	180 to 270 km

BUSINESS CASE

Manufacturing
Cost for particle
collector



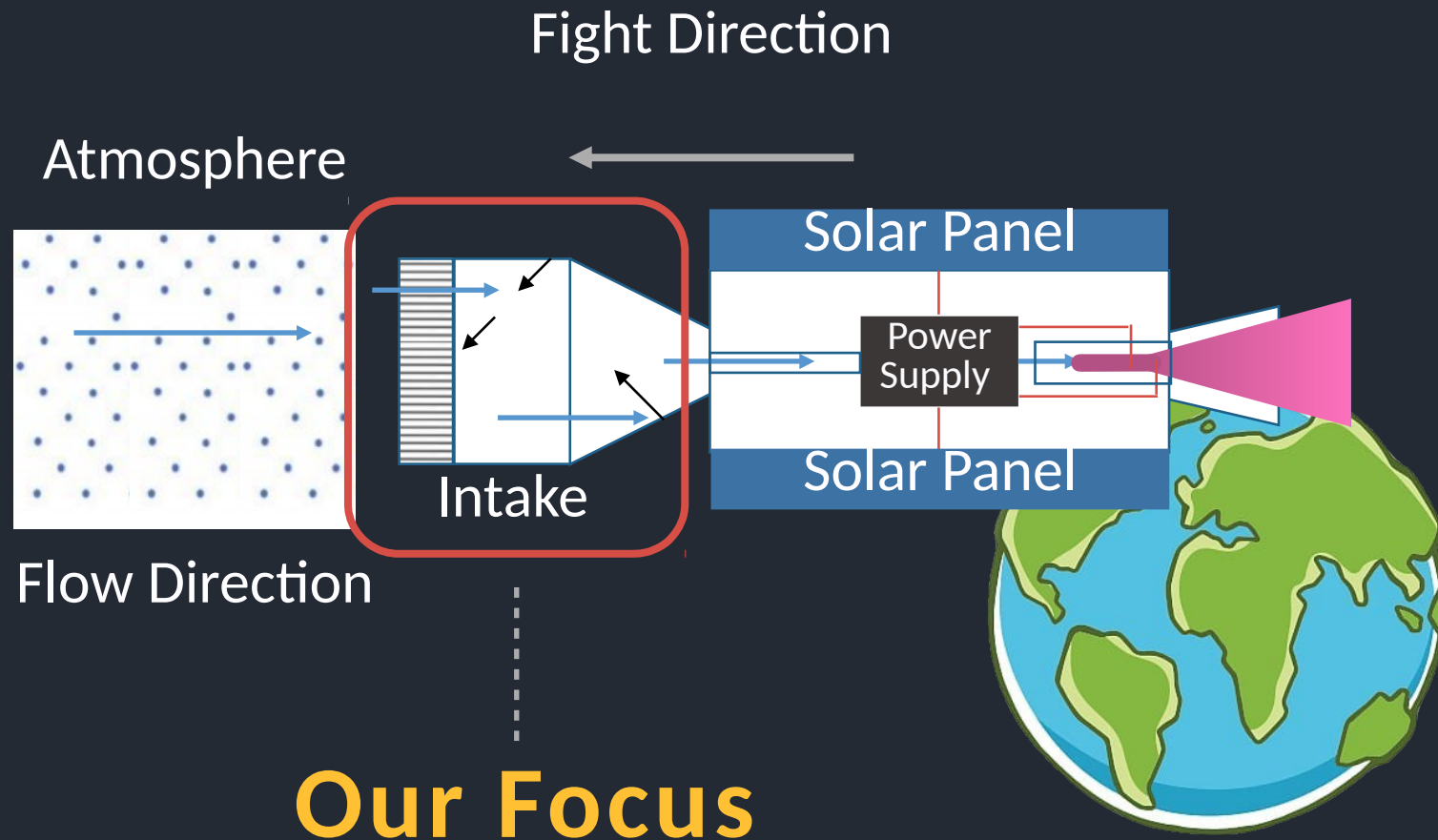
Cost vs Altitude for
Fixed Resolution & Coverage



SOLUTION details

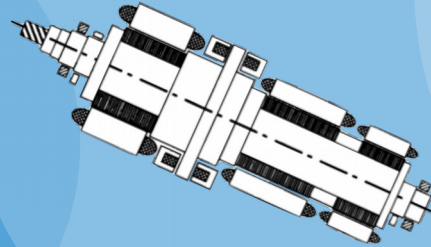
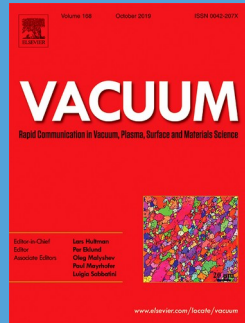


What is Air Breathing Electric Propulsion?



Scientific Validation

Magnetic Bearings used
in milling Machines

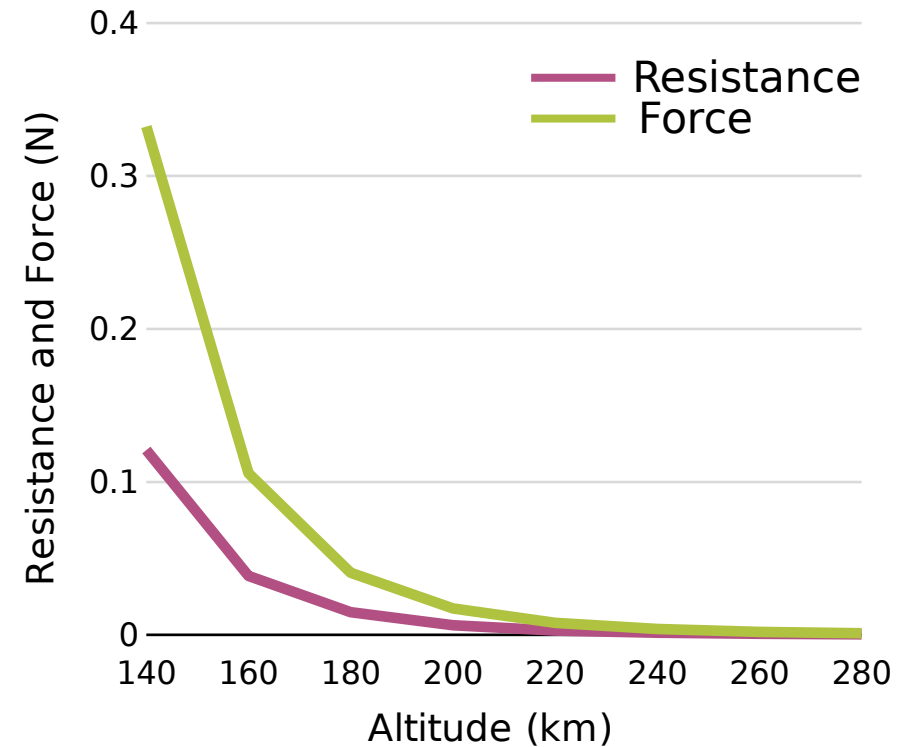
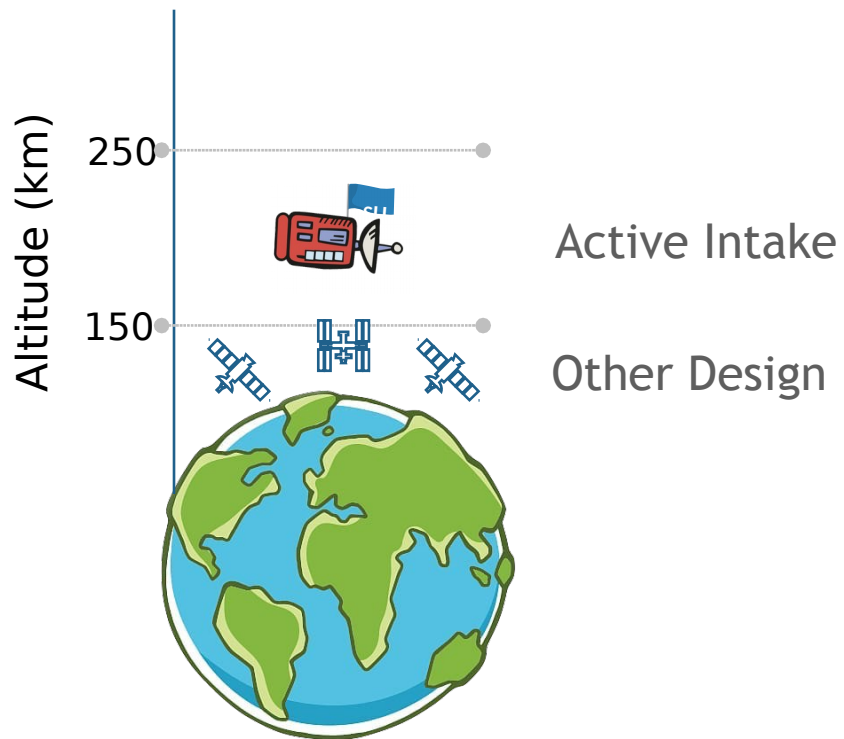


Active Intake Possibility to
reach 40 K Rpm
using Magnetic
Suspension

Magnetic
Suspension in
Space

Engineering Validation

Operational number density of
conventional 10 cm HET



Calculation based on
<https://ccmc.gsfc.nasa.gov/modelweb/models/nrlmsise00.php>

END USER FEEDBACK



"...you'd have unlimited force available. ... it has the potential to revolutionize the space industry"

Stefano Antonetti
Technical Team Lead
Sfantonetti@gmail.com



"It's a great option as you don't need to carry any fuel"

Anatoliy Kopik
Project Manager & Marketing Director
Anatoliy.Kopik@sputnix.ru

TsAGI 100

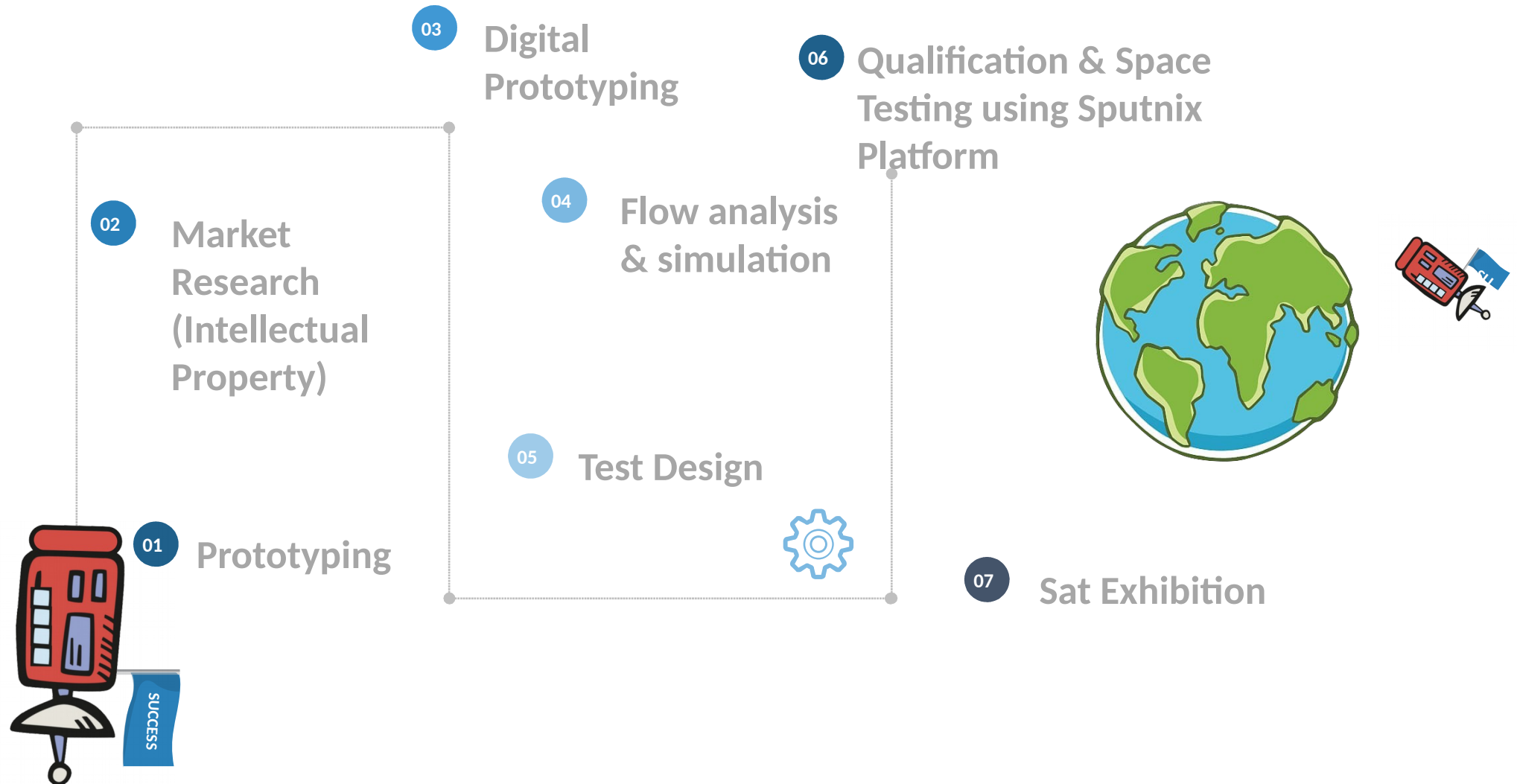
"Active intakes are usually heavy and not feasible, ...your solutions looks like a great idea"

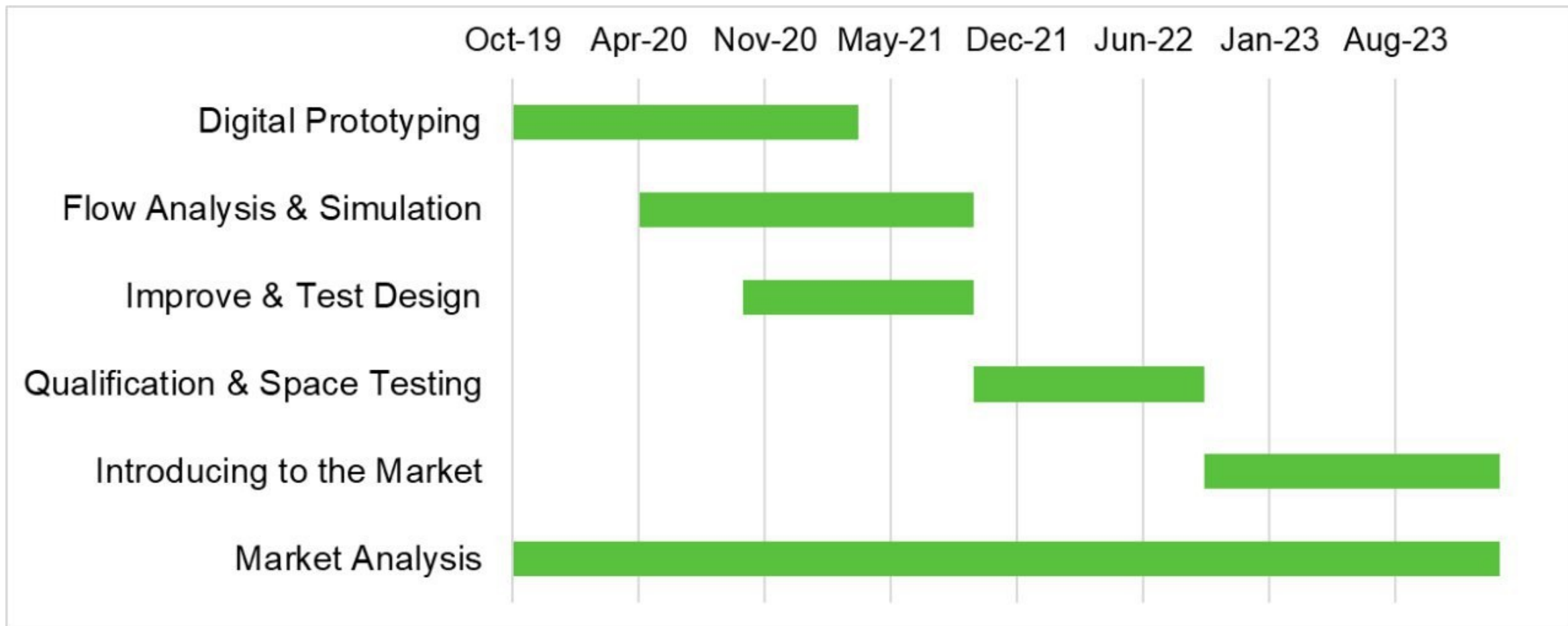
Alexander Golikov
Leading Scientist
agolikov@yandex.ru



***"You can use our launch platform,
We will send your working prototype to Space!"***

LONG TERM PLAN





Meet The Team



Salman Ali Thepdawala

Space Engineering



Ekaterina Trofimova

Data Science
Economics



Alexey Bunkov

Electronics
Quantum and Photonics



Lev Popyvanov

3d printing
Prototype Designe



Liliya Mironova

Advanced Manufacturing Technologies
Computer Science

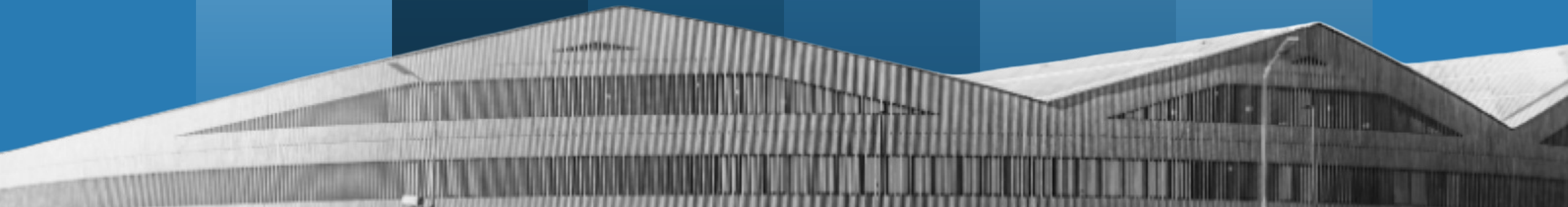


Iurii Lebedev

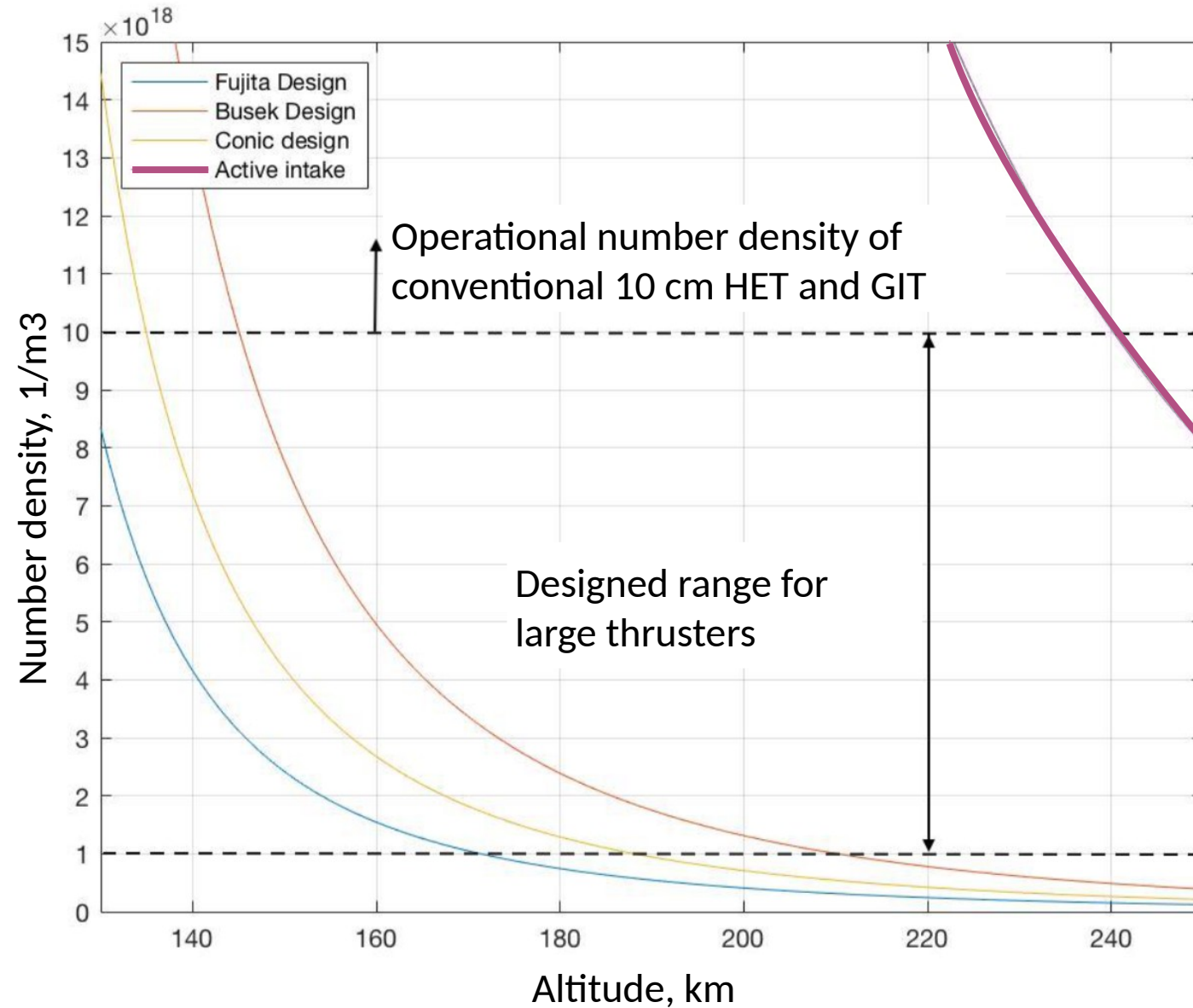
Records Manager



THANKS FOR WATCH



Appendix 1. Density dependence on Altitude



Appendix 2. Parameters

Altitude: 206 km

GSR = 100 cm

Geometry:

Diameter: 0.5 m

Length = 1.46 m

Solar Arrays Area = 1.9 m²

Body Drag = 3.77 mN ($C_d = 2.87$)

Solar Panel Drag = 1.65 mN ($C_d = 19.1$)

Overall Drag = 5.42 mN

Available Thrust = 6.9 mN