1. What is the value of the pressure coefficient C_p at a stagnation point when (a) $M_\infty = 0.7$ and (b) $M_\infty \to \infty$ for $\gamma = 1.4$?

2. A rectangular wing with a NACA 0006 airfoil section having an aspect ratio of 3.5 is flying at $M_\infty = 0.85$ at a height of 12 km. What is the airfoil section and the aspect ratio for an equivalent wing in an incompressible flow?