Horseshoe orbit :

In an rotating non inertial frame:

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The equations of motion are :
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 $rac{\overline{dt}}{dt} = rac{\overline{\mu}}{\mu}$ $d\phi = p_\phi$

 $rac{d\phi}{dt} = rac{p_\phi}{\mu r^2} - \omega$

 $rac{dp_r}{dt} = rac{p_{\phi}^2}{\mu r^3} - rac{Gm_1 \mu}{s_1^3} [r - r_1 \, \cos(\phi - \pi)] - rac{Gm_2 \mu}{s_2^3} [r - r_2 \, \cos(\phi)]$

 $rac{dp_{\phi}}{dt} = -rac{Gm_{1}\mu}{s_{1}^{3}}[rr_{1}\,\sin(\phi-\pi)] - rac{Gm_{2}\mu}{s_{2}^{3}}[rr_{2}\,\sin(\phi)]$

Here, $s_1^3=[r^2+r_1^2-2rr_1\,\cos(\phi-\pi)]^{3/2}$ and $s_2^3=[r^2+r_2^2-2rr_2\,\cos(\phi)]^{3/2}$

Now let's write the program:

```
import matplotlib.pyplot as plt
import seaborn as sns
import math as mth
import numpy as np
sns.set_style('darkgrid')
In [5]: # The veriebles will be read as an and so
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```
# The variables will be: r, o, pr and po
# Define the functions:
def s13(t,r,o):
   return np.sqrt(r**2 + r1**2 - 2.0*r*r1* np.cos(o - np.pi))**3
def s23(t,r,o):
   return np.sqrt(r^{**}2 + r2^{**}2 - 2.0^{*}r^{*}r2^{*} np.cos(0))**3
def fr(t,r,o,pr,po):
               \# \ u = 1
   return pr
def fo(t,r,o,pr,po):
   return (po/(r**2)) - w
def fpr(t,r,o,pr,po):
    def fpo(t,r,o,pr,po):
    #Define the constants:
rm = 1000 # m1/m2
w = 2.0* (np.pi/(27.0*24.0)) #rad/hour
r1 = (3.84e5)/(1+rm)
r2 = (3.84e5*rm)/(1+rm)
gm1 = w**2 * r2 * 14.7e10
gm2 = gm1/rm
# Define the initial constants:
tf = 24*1600 # hours
h = 0.1 # hours
n = int((tf-ti)/h)
r0 = 384000 \# (km)
o0 = np.pi # 180 degree
pr0 = 0
po0 = r0**2 * w
# start process :
t = ti
r = r0
0 = 00
pr = pr0
po = po0
x = r0* np.cos(00)
y = r0* np.sin(o0)
xe = r1* np.cos(np.pi)
ye = r1* np.sin(np.pi)
xm = -r2* np.cos(0)
ym = -r2* np.sin(0)
r_list = [r0]
o_list = [00]
pr_list = [pr0]
po_list = [po0]
t_list = [ti]
x_{list} = [r* np.cos(o)]
y_list = [r* np.sin(o)]
xe\_list = [r1* np.cos(np.pi)]
ye_list = [r1* np.sin(np.pi)]
xm_list = [-r2* np.cos(0)]
ym_list = [-r2* np.sin(0)]
for i in range(n):
   #print(t,r,o,pr,po,x,y,xe,ye,xm,ym)
   k1r = fr(t,r,o,pr,po)
   k10 = fo(t,r,o,pr,po)
   k1pr = fpr(t,r,o,pr,po)
   k1po = fpo(t,r,o,pr,po)
   k2r = fr(t+ h/2.0, r+ (k1r*h)/2.0, o+ (k1o*h)/2.0, pr+ (k1pr*h)/2.0, po+ (k1po*h)/2.0)
   k20 = fo(t+ h/2.0, r+ (k1r*h)/2.0, o+ (k1o*h)/2.0, pr+ (k1pr*h)/2.0, po+ (k1po*h)/2.0)
   k2pr = fpr(t+ h/2.0, r+ (k1r*h)/2.0, o+ (k1o*h)/2.0, pr+ (k1pr*h)/2.0, po+ (k1po*h)/2.0)
   k2po = fpo(t+ h/2.0, r + (k1r*h)/2.0, o+ (k1o*h)/2.0, pr+ (k1pr*h)/2.0, po+ (k1po*h)/2.0)
   k3r = fr(t+ h/2.0, r+ (k2r*h)/2.0, o+ (k2o*h)/2.0, pr+ (k2pr*h)/2.0, po+ (k2po*h)/2.0)
   k30 = fo(t+ h/2.0, r + (k2r*h)/2.0, o+ (k2o*h)/2.0, pr+ (k2pr*h)/2.0, po+ (k2po*h)/2.0)
   k3pr = fpr(t+ h/2.0, r+ (k2r*h)/2.0, o+ (k2o*h)/2.0, pr+ (k2pr*h)/2.0, po+ (k2po*h)/2.0)
   k3po = fpo(t+ h/2.0, r + (k2r*h)/2.0, o+ (k2o*h)/2.0, pr+ (k2pr*h)/2.0, po+ (k2po*h)/2.0)
   k4r = fr(t+h, r+ k3r*h, o+ k3o*h, pr+ k3pr*h, po+ k3po*h)
   k40 = fo(t+ h, r + k3r*h, o+ k3o*h, pr+ k3pr*h, po+ k3po*h)
   k4pr = fpr(t+ h, r+ k3r*h, o+ k3o*h, pr+ k3pr*h, po+ k3po*h)
   k4po = fpo(t+ h, r + k3r*h, o+ k3o*h, pr+ k3pr*h, po+ k3po*h)
   r = r + (h/6.0)*(k1r + 2.0* k2r + 2.0* k3r + k4r)
   0 = 0 + (h/6.0)*(k10 + 2.0* k20 + 2.0* k30 + k40)
   pr = pr + (h/6.0)*(k1pr + 2.0* k2pr + 2.0* k3pr + k4pr)
   po = po + (h/6.0)*(k1po + 2.0* k2po + 2.0* k3po + k4po)
   t = t+h
   x = r^* np.cos(0)
   y = r* np.sin(o)
   xe = r1* np.cos(w*t)
   ye = r1* np.sin(w*t)
   xm = -r2* np.cos(w*t)
   ym = -r2* np.sin(w*t)
   r_list.append(r)
   o_list.append(o*(180.0/mth.pi))
   pr_list.append(pr)
   po_list.append(po)
   x_{list.append(x)}
```

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In [8]: plt.plot(x_list, y_list)
    plt.rcParams["figure.figsize"] = (20, 10)
    plt.title("Trajectory of the Satellite")
    plt.xlabel("x")
    plt.ylabel("y")
    plt.plot(xe_list[0],ye_list[0], marker="o",markersize=20, color = 'green')
    plt.annotate("Earth",(xe_list[0],ye_list[0]))
    plt.plot(xm_list[0],ym_list[0], marker="o",markersize=10, color = 'blue')
    plt.annotate("Moon",(xm_list[0],ym_list[0]))
```

Out[8]: Text(-383616.3836163836, -0.0, 'Moon')

y_list.append(y)

