



Dat file NACA 2312 Airfoil M=2.0% P=30.0% T=12 ▲ 1.000000 -0.0000000 0.996090 0.000795 0.984418 0.003144 0.965159 0.006939 0.938603 0.012009 0.905153 0.018140 0.865318 0.025081 0.819709 0.032566 ▼