Student Satellite Project
Indian Institute of Technology, Bombay
Powai, Mumbai - 400076, INDIA

Website: www.aero.iitb.ac.in/satlab

Readme file for TLE2OrbitElements.py
Attitude Determination and Control Subsystem

Author: Sumit Agrawal
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This code is convert the given Two Line Element (TLE) into orbital elements.
Input: Two Line Element.
Do not copy paste the TLE from some other source like (n2yo.com). If you do make sure you rewrite the signs(+,-) as python sometimes do not understand signs(some syntex error).
Output:Orbital Elements {Inclination (degrees),Right ascension of the ascending node (degrees), Eccentricity, Argument of perigee (degrees) ,Mean Anomaly (degrees),Mean Motion (revolutions per day) }; First time derivative of mean motion; Second time derivative of mean motion; Bstar; Epoch year; Epoach and other parameter Satellite No., Class No., International designator

References: https://en.wikipedia.org/wiki/Two-line_element_set