

REF: WP-000001

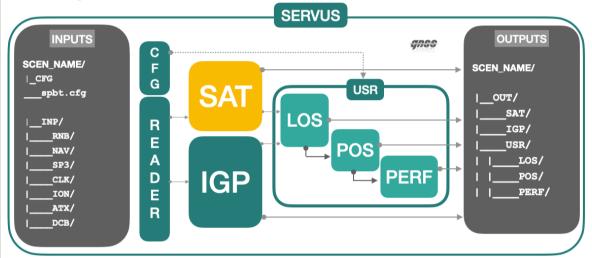
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PROJECT: SERVUS	WP Number: WP-0000001	
WP TITLE: WP1: Satellite	Issue: 1.0	
LEAD CONTRACTOR GNSS Academy		
CUSTOMER	ESA	WP LEADER: Student
ESTIMATED EFFORT	20 hours	
OR IECTIVE		

The main goal of this Work-Package is to develop the **Satellite Performance Module** of SERVUS tool by providing the daily satellite performance characterization on a EGNOS SIS real data campaign in January 2019

SCOPE

SERVUS high-level Architecture:



The **SERVUS SAT** module, the one in charge of computing SRE and the SigmaFLT as the main satellite indicators, is already implemented producing a daily file with the instantaneous information allowing to later characterize the satellite performance.

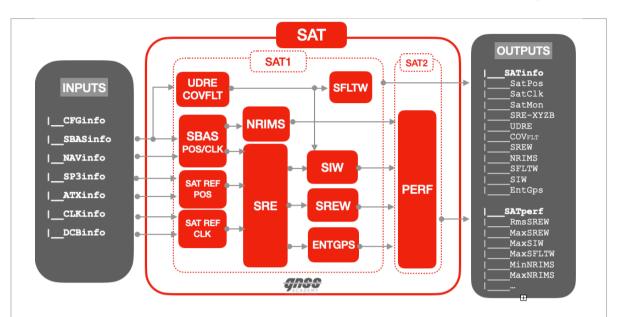
The main functions of this module are:

- Compute the Satellite Position and Clock from GPS Navigation Message
- Extract SBAS SIS satellite-related information:
 - o FLT Corrections: FC (MT2-5/24), LTC (MT24-M25)
 - o FLT Integrity Sigmas: UDRE (MT2-6), COV Matrix (MT27 or MT28)
 - Satellite Monitoring Flags (M/NM/DU) (MT2-6)
 - Degradation Factors (MT7-MT10)
- Compute the Satellite Corrected Position applying the corrections to the Navigation message.
- Compute True Orbit and Clock Information from reference files.
- Compute Satellite level main indicators:
 - o Orbit&Clock Error: SRE-XYZ and SRE-ACR and SRE-B -> SREW (@WUL)
 - o Error Variance: SigmaFLT at the WUL
 - o Satellite Safety Index: SREU/5.33 SigmaFLT at WUL
 - o Time Scale Offset: ENT-GPS OFFSET
- Assess the Satellite Performance in terms of accuracy (RMS SRE), Integrity (SI) and upperbound (SigmaFLT)



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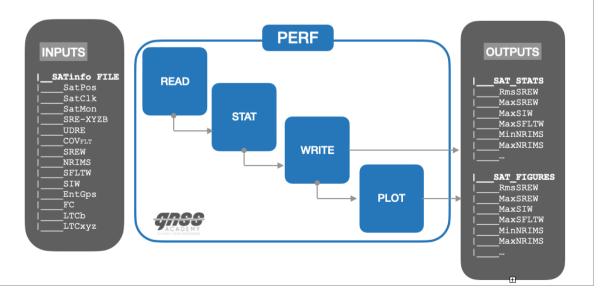


This first Work-package is related to the satellite-level characterization performances in terms of accuracy, integrity, and upper-bound.

- **Satellite Accuracy** is given by different statistics of the SRE (Satellite Residual Error) including the satellite orbit and clock errors.
- Satellite Upper-bounding and Continuity characterization is given by some statistics related to the SigmaFLT (equivalent to UDRE or SigmaUDRE)
- **Satellite Integrity** is given by the safety index SREU/5.33SigmaFLT at the Worst User location of the service are in order to ensure that the provided SigmaFLT correctly bounds the Error up to the required confidence levels.

This Work-package aims at developing the SAT2 Module corresponding to SAT PERF.

- 1. Read Satellite Instantaneous Information file.
- 2. Compute Satellite Statistics.
- 3. Write a Satellites Statistics file.
- 4. Plot Satellite Statistics.





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INPUTS

This Work-package uses daily input files as the output of the SAT module.

1. SAT_INFO_Y19D014_G123.dat (See Description Below)

FILENAME: SAT_INFO_Y19D014_G123_50s.dat

Column	Content	Format	Units	Description
C1	SOD	%5d	SEC	Second of Day
C2	DOY	%03d	DAYS	Day of the Year
С3	PRN	%s%02d	No Units	Satellite PRN for GPS, Galileo or GEO
C4	SAT-X	%f	KMS	Satellite Navigation Position (CURRENT) X-Component
C5	SAT-Y	%f	KMS	Satellite Navigation Position (CURRENT) Y-Component
C6	SAT-Z	%f	KMS	Satellite Navigation Position (CURRENT) Z- Component
C7	MONSTAT	%d	No Units	Satellite Monitoring Status (0: NM; 1:M; - 1; DU)
C8	SRESTAT	%d	No Units	Satellite SRE Status (0: NOK 1:OK)
С9	SREx	%f	METER	Satellite Residual Error X-Component
C10	SREy	%f	METER	Satellite Residual Error Y-Component
C11	SREz	%f	METER	Satellite Residual Error Z-Component
C12	SREb1	%f	METER	Preliminary Satellite Residual Error Clock Bias Component (containing the ENT-GPS Offset)
C13	SREW	%f	METER	Satellite Residual Error projected at the Worst User Location (WUL)
C14	SFLT-W	%f	METER	SigmaFLT/DFC projected at the Worst User Location (WUL)
C15	UDREI	%d		Satellite UDREI/DFREI Indicator
C16	FC	%f	METER	Satellite Clock Fast Corrections
C17	AF0	%f	METER	Satellite Clock Long Term Corrections Offset(LTCb=AF0+AF1(t-t0))=AF0 in EGNOS V2
C18	AF1	%f	METER	Satellite Clock Long Term Corrections drift This term is NULL in EGNOS V2
C19	LTCx	%f	METER	Satellite Orbit Long Term Corrections X-Component
C20	LTCy	%f	METER	Satellite Orbit Long Term Corrections Y-Component
C21	LTCz	%f	METER	Satellite Orbit Long Term Corrections Z-Component
C22	NRIMS	%d	-	Number of RIMS inside the ECAC that are in view from the satellite over a mask angle for, at least, 10min
C23	RDOP	%f	_	Radial DOP is the inverse DOP as seen from the Satellite to RIMS projected into the Radial direction

2. RIMS_REF_POSITIONS_2019.dat (See Description Below)

FILENAME: RIMS_REF_POSITIONS_2019.dat

Column	Content	Format	Units	Description
C1	RIMS _{FLAG}	%4d	none	Selection flag to activate or deactivate this RIMS from the PAN processing
C2	RIMS _{ACRONYM}	%8s	none	RIMS Acronym with 4 characters (e.g: ACRA)



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Canada

С3	RIMS _{ID}	%02d	none	RIMS identifier as an integer between 00 and 99.
C4	RIMS _{LON}	%15.7f	deg	RIMS geographical longitude positon
C5	RIMSLAT	%15.7f	deg	RIMS geographical latitude
С6	RIMS _{ALT}	%12.3f	m	RIMS height in meters
C7	RIMS _{MASK}	%10.2f	deg	RIMS Mask angle
C8	RIMSACQ	%5d	s	RIMS Receiver Acquisition time
С9	RIMS _{SITE}	%20s	none	RIMS site location as the representative name of the city
C10	RIMS _{COUNT}	%20s	none	RIMS Country location

RIMS Parameters

Selection flag [0:OFF/1:ON]

Station Name Acronym [%4s] Station Number ID [%2d]

Longitude [deg] Latitude [deg] Height [meters]

p7: Mask Angle [deg] # p8: Acquisition Time [minutes] # p9: Site [%s] # p10: Country [%s] | p4 | p5 # p1 | p2 | p3 | | p6 | p8 | p9 | 31.6168738 -28.6228066 22.3584909 38.5186833 Abu-Simbel Egypt Azores -28.6228066 9.8831919 29.9557435 23.7911121 13.5410936 -15.3793852 198.582 42.678 12.531 368.684 83.432 51.123 201.929 47.297 36.873 89.528 64.261 ALBA 57.0963560 10.00 Alborg Alexandria Denmark Egypt Greece Switzerland 31 1829123 AT.YA 37.8612226 52.3569463 27.9454109 51.8446852 BRNA CNRA Spain -8.4962089 15.0619850 10.7717472 -14.4007338 Ireland CRKA Italy Tunisia Island CTNA DJAA 37.4636507 33.8801909 65.2800856 55.8706262 10.771/47/2
-14.4007338
-4.4456436
32.8035793
17.1331590
-8.4915201
29.9313365
27.5553483
-17.7603716
-9.1298707
15.3965961
-16.7056705
-4.5144494
2.3349038
2.7387532
-21.9315020
12.5837594
-8.4248159
-1.2869796
1.4970244
10.8993146 Ireland GLGA 64.261 55.8706262 39.6381446 60.6663660 70.9728644 69.7311459 60.5388860 28.6196424 38.7887635 78.2285265 64.261 1088.564 53.299 63.463 119.770 35.773 137.719 157.600 496.566 GOLA GVLA JMEA Turkey Sweeden Norway KIRA Norway LAPA Finland LPIA LSBA LYRA Spain Portugal Norway 496.566 192.855 60.881 123.380 51.540 83.945 172.877 437.005 611.360 76.428 253.477 Portugal Spain France MADA 32.7474268 36.6846957 MT.GA 48.8359007 39.5636956 64.1292866 PARA PDMA Spain Island 25 26 27 28 29 30 41.8039468 42.9264046 42.6848906 50.8871205 ROMA Italv Spain Bulgaria England TLSA 43.4285678 France 253.477 43.875 136.126 125.452 514.795 35.278 239.597 119.179 1548.973 10.8993146 18.9390987 21.0676369 8.5648888 63.4568329 69.6631570 52.2147521 47.4537484 TRDA Norway Switzerland ZURA 8.5648888 -15.9544648 35.0220640 -9.4115837 27.7071958 -64.7692785 NOUA 18.0818926 Mauritania 32.7789087 30.3185120 -25.8833373 Israel Morocco SouthAfrica HFAA

31.816

OUTPUTS

Two main kind of outputs for this Work Package are required:

46.0711908

A Daily file with per satellite performance Statistics with the following format and content: SAT_STAT_Y19D014_G123_50s.dat

Column				Description
C1	PRN	%s	No Units	Satellite PRN for GPS (e.g: G01)
C2	MON	%6.2f	%	Satellite Monitoring Percentage along the day
С3	RIMS-MIN	%4d	No Units	Minimum Number of RIMS in view during the day
C4	RIMS-MAX	%6d	No Units	Maximum Number of RIMS in view during the day
C5	SREaRMS	%10.3f	METER	RMS of the SRE-A along the day, the Satellite Orbit Residual Error Along-Track component



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SRECRMS	%8.3f	METER	RMS of the SRE-C along the day, the Satellite Orbit Residual Error Cross-Track component
SRETRMS	%8.3f	METER	RMS of the SRE-R along the day, the Satellite Orbit Residual Error Radial component
SREBRMS	%8.3f	METER	RMS of the SRE-B along the day: the Satellite Residual Error Clock component SRE-B is computed subtracting the ENT-GPS Offset from the SREb1
SREWRMS	%8.3f	METER	RMS of SREW along the day
SREWMAX	%8.3f	METER	Maximum SREW reached along the day
SFLTMAX	%8.3f	METER	Maximum SigmaFLT@WUL reached along the day
SFLTMIN	%8.3f	METER	Minium SigmaFLT@WUL reached along the day
SIMAX	%8.3f	No Units	Maximum SI_{SAT} reached along the day $SI_{SAT} = \frac{SREW}{5.33 \cdot SFLT_W}$
FCMAX	%8.3f	METER	Maximum Satellite Clock Fast reached along the day.
LTCbMAX	%8.3f	METER	Maximum Clock Long Term Corrections (LTCb) reached along the day
LTCxMAX	%8.3f	METER	Maximum Satellite Long Term Corrections X- Component (LTCx) reached along the day
LTCyMAX	%8.3f	METER	Maximum Satellite Long Term Corrections Y- Component (LTCy) reached along the day
LTCzMAX	%8.3f	METER	Maximum Satellite Long Term Corrections Z- Component (LTCz) reached along the day
NMI	%8.3f	No Units	Number of Misleading Information (MI). Number of times SIW > 1
NTRANS	%4d	No Units	Number of Transitions from M to NM and M to DU
	SRETRMS SREWRMS SREWRMS SREWMAX SFLTMAX SFLTMIN SIMAX LTCDMAX LTCDMAX LTCMAX LTCYMAX LTCYMAX LTCZMAX NMI	SRETRMS %8.3f SREBRMS %8.3f SREWRMS %8.3f SREWMAX %8.3f SFLTMAX %8.3f SFLTMIN %8.3f SIMAX %8.3f LTCDMAX %8.3f LTCXMAX %8.3f LTCYMAX %8.3f LTCZMAX %8.3f LTCZMAX %8.3f LTCZMAX %8.3f NMI %8.3f	SRExRMS %8.3f METER SREbRMS %8.3f METER SREWRMS %8.3f METER SREWMAX %8.3f METER SFLTMAX %8.3f METER SFLTMIN %8.3f METER SIMAX %8.3f METER LTCMAX %8.3f METER LTCxMAX %8.3f METER LTCyMAX %8.3f METER LTCzMAX %8.3f METER NMI %8.3f No Units

• A Daily file containing the estimated ENT-GPS per epoch: ENTGPS_Y19D014_G123_50s.dat

Column	Content	Format	Units	Description
C1	SOD	%5d	SEC	Second of Day
				ENT to GPS Offset computed as follows:
				$ENTGPS = median(\{SREb1 - SREr\}_{SRE_{STATUS}==1})$
C2	ENT-GPS	%10.4f	SEC	Where SREr is the projection of (SREx, SREy, SREz) into the satellite radial direction
				And SREb1 is the preliminary Satellite Clock Error including the time scales offset

• Daily Satellite performance plots

TASKS		
ID	DESCRIPTION	
T0. PRELIMINARY	Downloading and Understanding	
T0.1 FOLDER TREE	Download following TAR file:	
	STEP1. Download SERVUS-TOOL folder tree and files	
	→ SBPT-SERVUS_WP1.tgz	
	STEP2. Place it somewhere in the GNSS Academy working directory	
	STEP3 untar the file.	



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	tar xvfz SBPT-SERVUS_WP1.tgz	
	Check that all the following information is available:	
	SBPT/SERVUS/SERVUS_V1.0/SERVUS_WP1_SAT/ SRC	
	SatPerformances.py	
	SatFunctions.py COMMON	
	PlotsConstants.py	
	Plots.py	
	Coordinates.py GnssConstants.py	
	Dates.py	
	SCN	
	SCEN-EGNOS-SIS-GEO123-JAN19	
	CFG	
	satperformances.cfg	
	INP	
	NAV/brdc0140.19n RNB/M1230140.19b	
	EMS/GEO123/2019/DOY014/h*.ems	
	SP3 /igs20361.sp3	
	DCB/P1C11901.DCB & P1P21901.DCB	
	ION /igsg0140.19i	
	CLK/igs20361.clk_30s RIMS/RIMS REF POSITIONS 2019.dat	
	RIMS/RIMS_REF_FOSITIONS_2019.dat	
	OUT	
	SAT	
	Note that there are 3 SAT_INFO files already sampled at 50	
	seconds in order to speed-up the execution process.	
	Output Files will be generated in:	
	OUT/SAT/SAT_STAT_Y19D014_G123_50s.dat OUT/SAT/SAT_STAT_Y19D015_G123_50s.dat	
	OUT/SAT/SAT_STAT_Y19D015_G123_50s.dat	
	Output Figures shall be generated in:	
	OUT/SAT/figures/*.png	
T1. IMPLEMENTATION	Functions Implementation: Open/Reading/Loading	
T1.1 Configuration	Check configuration file. 3 configuration parameters, INI_DATE,	
	END_DATE (in order to run only one day, both shall be the same) and TSTEP.	
	CFG/satperformances.cfg	
T1.2 Execution	Execute SatPerformance Module	
	Call Main function from the SRC directory and check that you generate	



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	<pre>python SatPerformances.py/SCN/SCEN-EGNOS-SIS-GE0123-JAN19</pre>	
	Check that this runs properly and that already generates the following files: OUT/SAT/SAT_STAT_Y19D014_G123_50s.dat OUT/SAT/ENTGPS_Y19D014_G123_50s.dat	
	These files are partial files that have to be completed with the remaining columns.	
T1.3 Implementation	Implement the necessary Python functions to create the satellite statistics file with all the different columns, in line with following tasks.	
T2. ENTGPS	Create a file including the instantaneous ENT-GPS Offset	
SOD	Seconds of Day	
ENTGPS	ENT-GPS Offset: estimated from the SRE of the monitored satellites with SRE_STATUS OK, as follows:	
	$ENTGPS = median(\{SREb1 - SREr\}_{SRE_{STATUS}==1})$	
	Where SREr is the projection of (SREx, SREy, SREz) into the satellite radial direction	
T3. STATISTICS	Create a file including all satellite Statistics with the following	
PRN	information per columns. SAT_STATISTICS_ PRN: Satellite PRN	
	Titiv. Satellite Titiv	
MON	Satellite Monitoring Percentage in the day. Count the number of epochs the satellite has been monitored during the day divided by the total number of seconds in the day.	
MIN RIMS	Minimum number of RIMS in view by the satellite during the day Note that the satellite is monitored by EGNOS when the satellite is seen at least by 3 RIMS.	
MAX RIMS	Maximum number of RIMS in view by the satellite during the day	
RMS-SRE _R	Root Mean Square of Satellite Orbit Error - Radial Component	
	Note that SRE _R is not in the input file. It has to be computed by projecting the SRE _{XYZ} along the orbit radial component.	
	Radial Direction $U_R = r_s / r_s $	
RMS-SRE _C	Root Mean Square of Satellite Orbit Error - Cross Track Component	
	Note that SRE _C is not in the input file. It has to be computed by projecting the SRE _{XYZ} along the orbit cross track component.	
	Uc=UR X Uv	
	where	
	$U_R=r_s/ r_s $ in the direction of the satellite position (r_s) $U_V=v_s/ v_s $ in the direction of the satellite velocity (v_s)	
	Note that Velocity is obtained by deriving the Satellite Position during the monitoring periods. Care shall be taken for the first epoch of the satellite monitoring pass since	



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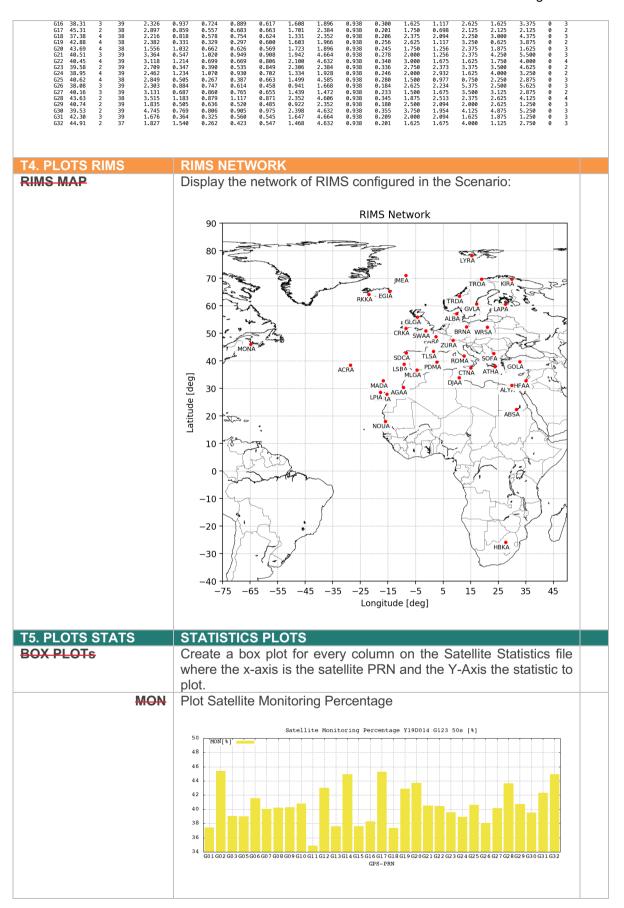
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	no previous position to compute the velocity. First sample will be discarded from the statistics.		
	Do not forget that it is a non-inertial reference frame and you need to add the Earth's Rotation in order to get the absolute velocity.		
	$V_{sat} = \frac{P_{sat}(k) - P_{sat}(k-1)}{\Delta t} + \Omega_{EARTH} x P_{sat}(k)$		
	# Earth's rotation rate (rad/sec) OMEGA_EARTH = 7.2921151467e-5		
RMS-SREA	Root Mean Square of Satellite Orbit Error - Along Track Component		
	Note that SRE _A is not in the input file. It has to be computed by projecting the SRE _{XYZ} along the orbit along track component.		
	Along-Track is the cross product between Cross and Radial Track Component. UA=Uc x UR		
	$RMS = \sqrt{\frac{\Sigma(X_k)^2}{N}}$		
RMS-SRE _B	Root Mean Square of SRE-B Clock Component SRE-B is computed subtracting the ENT-GPS Offset from the SREb1		
RMS-SREW	Root Mean Square of SREW: satellite Orbit Error at the Worst User		
MAX-SREW	Location Maximum SREW Satellite Orbit Error at the Worst User Location		
MAX SFLT	Maximum Sigma FLT as the Satellite Orbit and clock Error Sigma at the Worst User Location.		
MAX SIW	Maximum Satellite Safety Index as the ratio between the SRE and the SigmaFLT at the Worst User Location The instantaneous SI for each satellite shall be computed as follows: $SI_{SAT} = \frac{SREW}{5.33 \cdot SFLT_W}$		
MAX LTCx	Maximum (absolute)Value of the Satellite Position LTC correction		
MAX LTCy	X-Component Maximum (absolute)Value of the Satellite Position correction LTC Y-Component		
MAX LTCz	Maximum (absolute)Value of the Satellite Position correction LTC Z-Component		
MAX LTCb	Maximum (absolute) Value of the Satellite LTC -Clock AF0		
MAX FC	Maximum (absolute) Value of the Satellite Clock Fast Corrections		
NMI	Number of Satellite MIs (Misleading Information) as the number of times SI is higher than 1. (SI>1)		
NTRANS	Number of transitions Monitored to Not Monitored (MtoNM) or to Don't USE (MtoDU)		
SAT STAT Y19D014 G123 50s			
#PRN MON minRIMS MaxRIMS SREARMS SI G01 37.44 4 39 2.663 G02 45.43 4 38 2.739 G03 39.06 4 38 3.898 G05 39.00 3 39 1.421 G06 41.55 3 39 2.587 G07 40.05 5 39 2.397 G08 40.23 3 39 4.237 G09 40.26 3 39 4.227 G11 34.90 5 39 2.402 G12 43.00 1 39 4.173 G13 37.62 4 39 3.031 G14 44.91 4 38 1.675	REPRIS SREPRINS SPERRINS SREMANX SFLITMAX SFLITMIN SIMAX FGMAX LTC.MAX LTC.MAX LTC.MAX NNI NITANIS 0.591 0.488 0.470 0.562 1.235 2.384 0.938 0.228 1.750 1.675 2.375 3.625 4.580 0 3 0.581 0.428 0.352 0.630 2.214 2.352 0.938 0.228 1.750 0.977 2.375 3.625 4.580 0 3 0.947 0.691 0.587 0.482 0.352 0.630 2.214 2.352 0.938 0.228 1.750 0.977 2.375 3.625 4.580 0 3 0.949 0.580 0.948 0.586 0.948 0.275 0.948 0.9		



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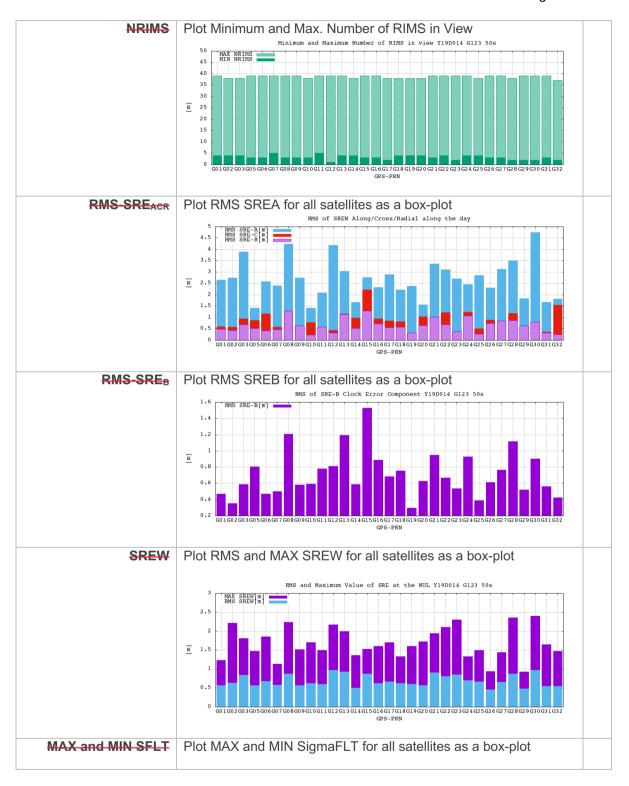
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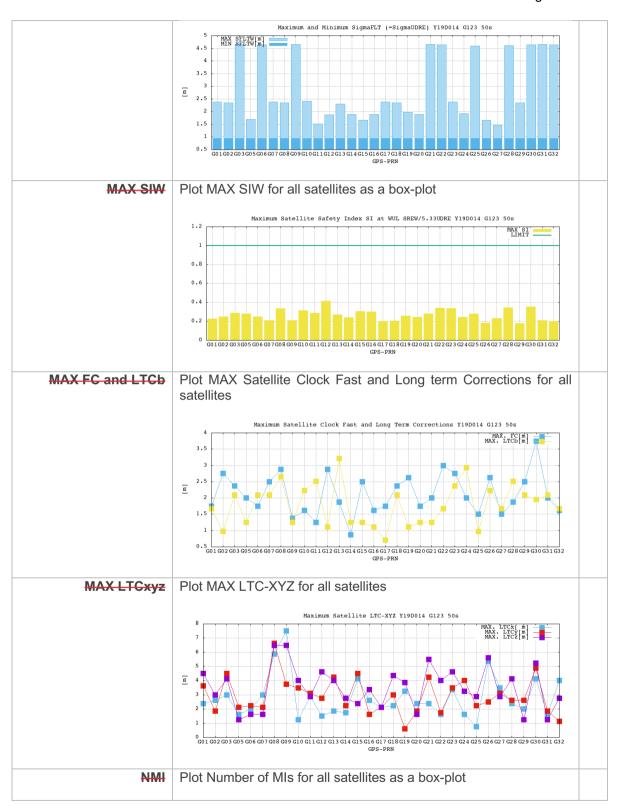
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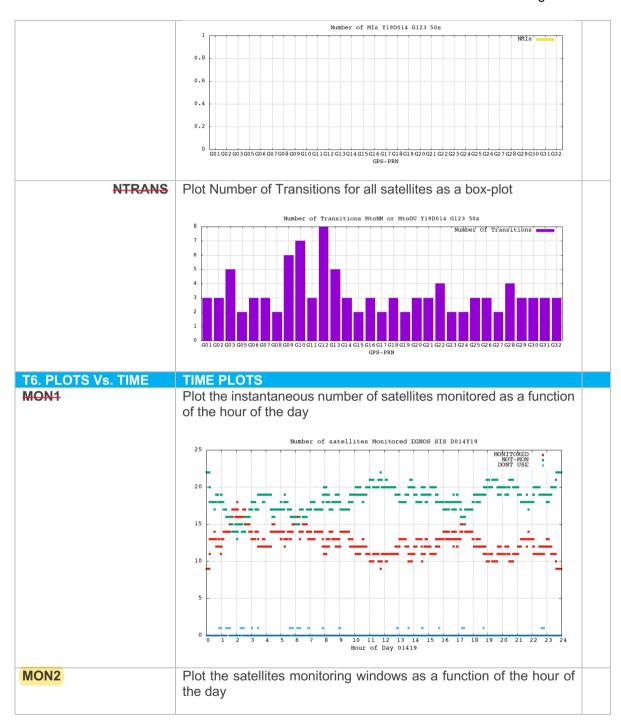
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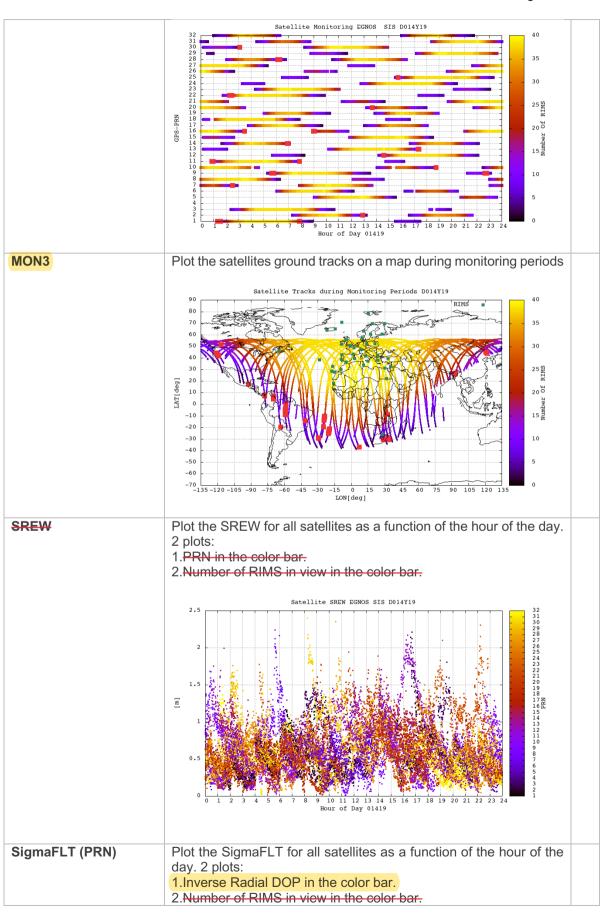
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