

PRELIMINARY



747-8 Airplane Characteristics for Airport Planning



Boeing Commercial Airplanes

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747-8 AIRPLANE CHARACTERISTICS

LIST OF ACTIVE PAGES

MARCH 2011 iii

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PRELIMINARY

1.0 SCOPE AND INTRODUCTION

1.1 Scope

1.2 Introduction

1.3 A Brief Description of the 747-8

PRELIMINARY

1.0 SCOPE AND INTRODUCTION

1.1 Scope

This document provides, in a standardized format, airplane characteristics data for general airport planning. Since operational practices vary among airlines, specific data should be coordinated with the using airlines prior to facility design. Boeing Commercial Airplanes should be contacted for any additional information required.

Content of the document reflects the results of a coordinated effort by representatives from the following organizations:

- Aerospace Industries Association
- Airports Council International – North America
- International Industry Working Group
- International Air Transport Association

The airport planner may also want to consider the information presented in the "Commercial Aircraft Design Characteristics - Trends and Growth Projections," for long range planning needs and can be accessed via the following web site:

<http://www.boeing.com/airports>

The document is updated periodically and represents the coordinated efforts of the following organizations regarding future aircraft growth trends.

- International Civil Aviation Organization
- International Coordinating Council of Aerospace Industries Associations
- Airports Council International – North America and World Organizations
- International Industry Working Group
- International Air Transport Association

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1.2 Introduction

This document conforms to NAS 3601. It provides characteristics of the Boeing Model 747-8 Freighter airplane for airport planners and operators, airlines, architectural and engineering consultant organizations, and other interested industry agencies. Airplane changes and available options may alter model characteristics. The data presented herein will be mainly the 747-8F at this time. As the data for the 747-8 Intercontinental becomes available for publication, it will be added. These data will reflect typical airplanes in each model category. Data used is generic in scope and not customer-specific.

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Fax: 425-237-2665

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1.3 A Brief Description of the 747-8

The 747-8 is the latest derivative of the 747 family of airplanes and is being developed in both Freighter and Passenger versions. The -8 is externally similar to the 747-400 with a higher gross weight, longer fuselage and increased wingspan. The 747-8 Freighter retains the 747-400F nose cargo door, continuing the capability to easily load outsized cargo. The 747-8 will use the new high bypass ratio engines which are the quiet, efficient GENx engines being developed for the 787 aircraft. By combining changes in the wing, adding the raked wingtips, and using the GENx engines, the 747-8 will be the fastest flying aircraft at Mach 0.845 for the Freighter and Mach 0.855 for the Intercontinental. The 747-8F will enter into service in 2011.

Other characteristics unique to the 747-8 compared to the 747-400 include:

- Next generation advanced alloys
- New wing design, including new airfoils and raked wingtips replacing the winglets
- 787 Next Gen engines, including light weight composite fan case and fan blades, modified to provide current 747 bleed requirements.
- Improved flight deck while preserving 747-400 operational commonality
- New interior architecture to enhance passenger experience
- Improved aerodynamic efficiency and reduced seat-mile cost (Passenger variant) and reduced ton-mile cost (Freighter variant)

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2.0 AIRPLANE DESCRIPTION

2.1 General Characteristics

2.2 General Dimensions

2.3 Ground Clearances

2.4 Interior Arrangements

2.5 Cabin Cross Sections

2.6 Lower Cargo Compartments

2.7 Door Clearances

PRELIMINARY

2.0 AIRPLANE DESCRIPTION

2.1 General Characteristics

Maximum Design Taxi Weight (MTW). Maximum weight for ground maneuver as limited by aircraft strength and airworthiness requirements. (It includes weight of taxi and run-up fuel.)

Maximum Design Takeoff Weight (MTOW). Maximum weight for takeoff as limited by aircraft strength and airworthiness requirements. (This is the maximum weight at start of the takeoff run.)

Maximum Design Landing Weight (MLW). Maximum weight for landing as limited by aircraft strength and airworthiness requirements.

Maximum Design Zero Fuel Weight (MZFW). Maximum weight allowed before usable fuel and other specified usable agents must be loaded in defined sections of the aircraft as limited by strength and airworthiness requirements.

Operating Empty Weight (OEW). Weight of structure, powerplant, furnishing systems, unusable fuel and other unusable propulsion agents, and other items of equipment that are considered an integral part of a particular airplane configuration. Also included are certain standard items, personnel, equipment, and supplies necessary for full operations, excluding usable fuel and payload.

Maximum Payload. Maximum design zero fuel weight minus operational empty weight.

Maximum Seating Capacity. The maximum number of passengers specifically certificated or anticipated for certification.

Maximum Cargo Volume. The maximum space available for cargo.

Usable Fuel. Fuel available for aircraft propulsion.

PRELIMINARY

CHARACTERISTICS	UNITS	GE _{nx} 2B67 ENGINES
MAX DESIGN TAXI WEIGHT	POUNDS	978,000
	KILOGRAMS	443,614
MAX DESIGN TAKEOFF WEIGHT	POUNDS	975,000
	KILOGRAMS	442,253
MAX DESIGN LANDING WEIGHT	POUNDS	757,000
	KILOGRAMS	343,370
MAX DESIGN ZERO FUEL WEIGHT	POUNDS	717,000
	KILOGRAMS	325,226
SPEC OPERATING EMPTY WEIGHT	POUNDS	TBD
	KILOGRAMS	TBD
MAX STRUCTURAL PAYLOAD	POUNDS	295,200
	KILOGRAMS	133,901
TYPICAL CARGO – MAIN DECK CONTAINERS	CUBIC FEET	24,462
	CUBIC METERS	693
MAX CARGO - LOWER DECK CONTAINERS (LD-1)	CUBIC FEET	5,850
	CUBIC METERS	166
MAX CARGO - LOWER DECK BULK CARGO	CUBIC FEET	520
	CUBIC METERS	14.7
USABLE FUEL CAPACITY	U.S. GALLONS	60,211
	LITERS	229,980
	POUNDS	407,045
	KILOGRAMS	184,628

NOTES:

1. SPEC OPERATING EMPTY WEIGHT REFLECTS STANDARD ITEM ALLOWANCES. ACTUAL OEW WILL VARY WITH AIRPLANE CONFIGURATION. CONSULT USING AIRLINE FOR ACTUAL OEW.

2.1.1 GENERAL CHARACTERISTICS MODEL 747-8F

PRELIMINARY

CHARACTERISTICS	UNITS	GEnx 2B67 ENGINES
MAX DESIGN TAXI WEIGHT	POUNDS	978,000
	KILOGRAMS	443,614
MAX DESIGN TAKEOFF WEIGHT	POUNDS	975,000
	KILOGRAMS	442,253
MAX DESIGN LANDING WEIGHT	POUNDS	682,000
	KILOGRAMS	309,350
MAX DESIGN ZERO FUEL WEIGHT	POUNDS	642,000
	KILOGRAMS	291,206
SPEC OPERATING EMPTY WEIGHT	POUNDS	TBD
	KILOGRAMS	TBD
MAX STRUCTURAL PAYLOAD	POUNDS	TBD
	KILOGRAMS	TBD
TYPICAL SEATING CAPACITY (INCLUDES UPPER DECK)	UPPER DECK	48 BUSINESS CLASS
	MAIN DECK	26 FIRST, 89 BUSINESS, 352 ECONOMY
MAX CARGO - LOWER DECK CONTAINERS (LD-1)	CUBIC FEET	6,345
	CUBIC METERS	180
MAX CARGO - LOWER DECK BULK CARGO	CUBIC FEET	640
	CUBIC METERS	18
USABLE FUEL CAPACITY	U.S. GALLONS	64,055
	LITERS	242,475
	POUNDS	429,159
	KILOGRAMS	194,659

NOTES:

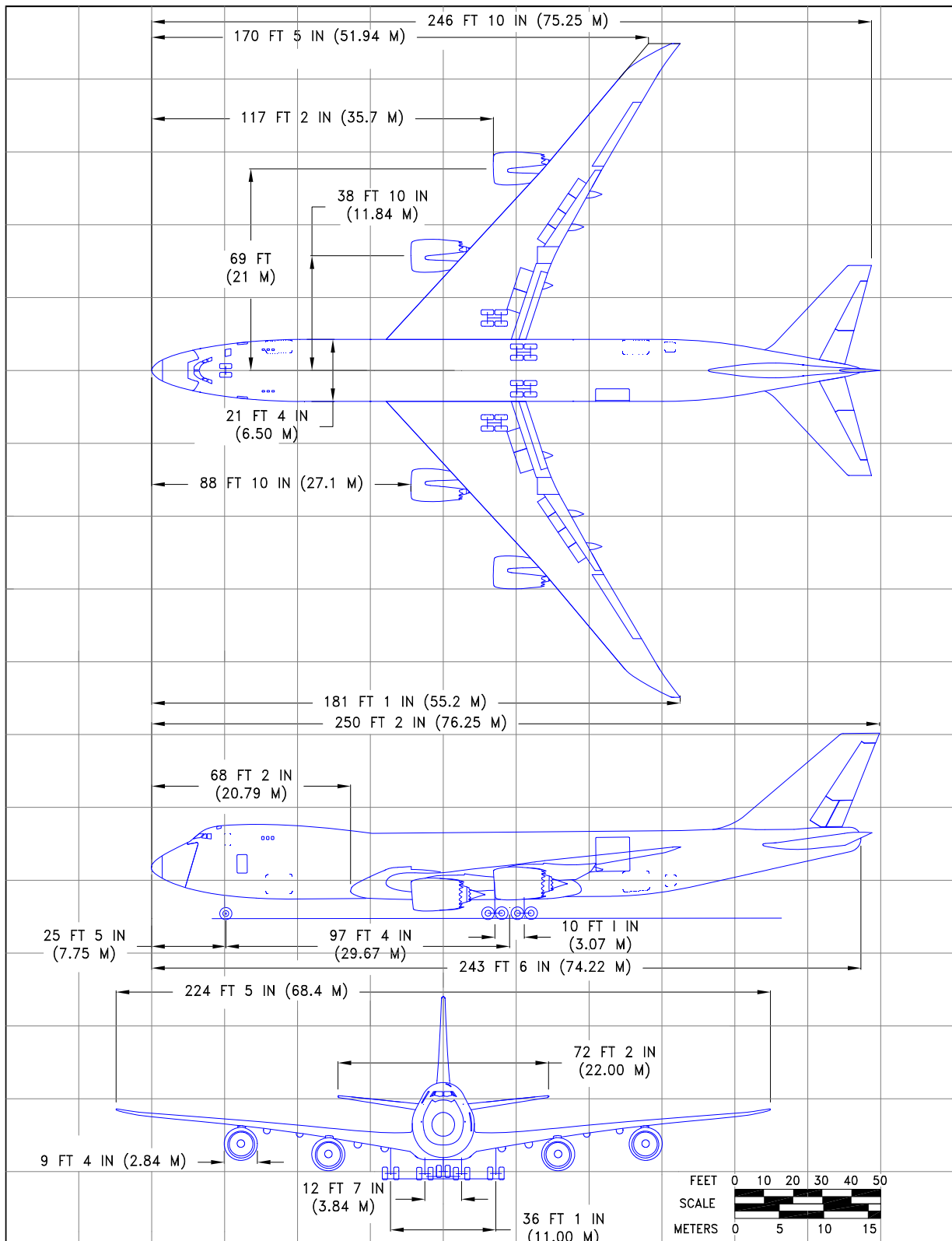
1. SPEC OPERATING EMPTY WEIGHT REFLECTS STANDARD ITEM ALLOWANCES. ACTUAL OEW WILL VARY WITH AIRPLANE CONFIGURATION. CONSULT USING AIRLINE FOR ACTUAL OEW.

2.1.2 GENERAL CHARACTERISTICS

MODEL 747-8

D6-58326-3

PRELIMINARY

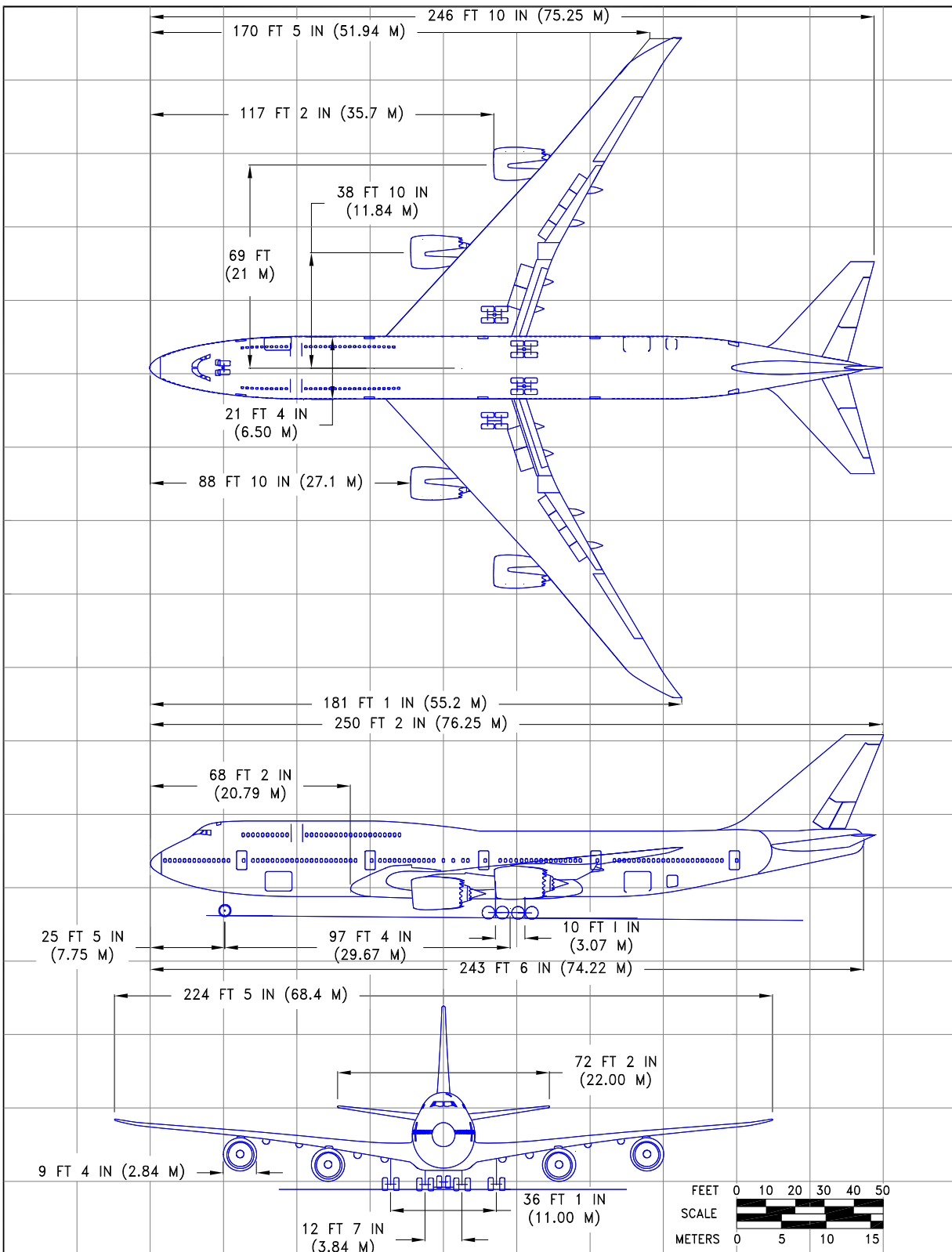


2.2.1 GENERAL DIMENSIONS MODEL 747-8F

D6-58326-3

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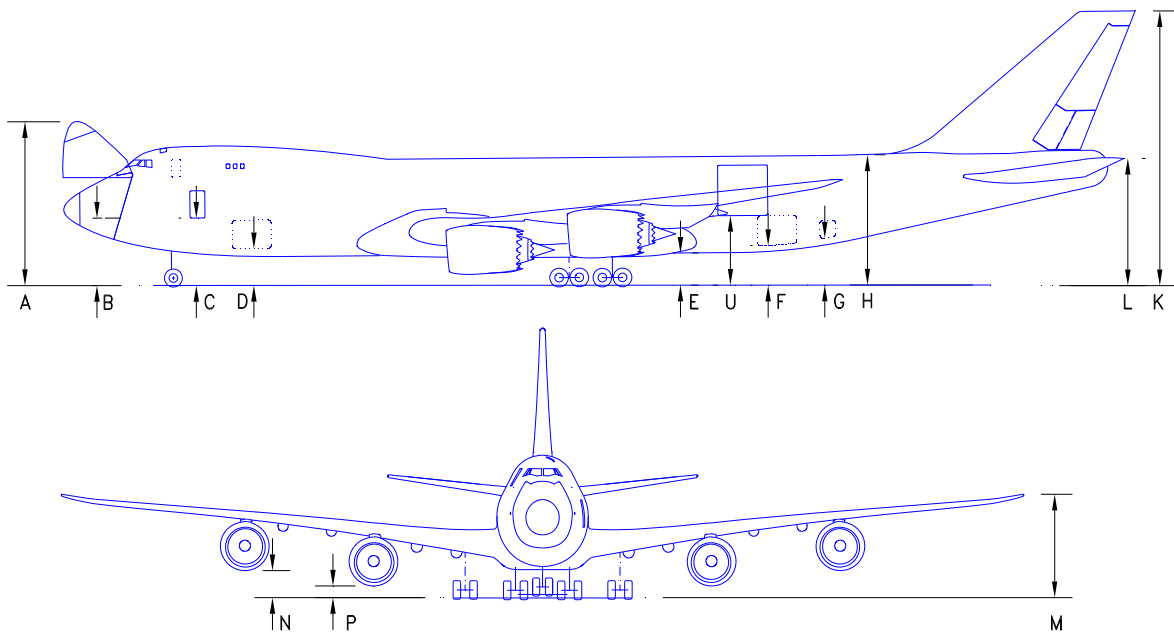
PRELIMINARY



2.2.2 GENERAL DIMENSIONS MODEL 747-8

D6-58326-3

PRELIMINARY



	MINIMUM		MAXIMUM	
	FT - IN	M	FT - IN	M
A	38 - 8	11.79	40 - 2	12.24
B	15 - 6	4.72	17 - 11	5.46
C	15 - 9	4.80	17 - 9	5.41
D	9 - 1	2.77	10 - 10	3.30
E	6 - 9	2.06	8 - 1	2.46
F	9 - 1	2.77	10 - 7	3.23
G	9 - 7	2.92	11 - 4	3.45
H	28 - 8	8.74	30 - 7	9.32
K	61 - 11	18.87	62 - 8	19.1
L	27 - 10	8.48	28 - 8	8.74
M	18 - 7	5.66	20 - 4	6.20
N	5 - 0	1.52	6 - 1	1.85
P	2 - 2	0.66	3 - 1	0.94
U	16 - 0	4.88	17 - 3	5.26

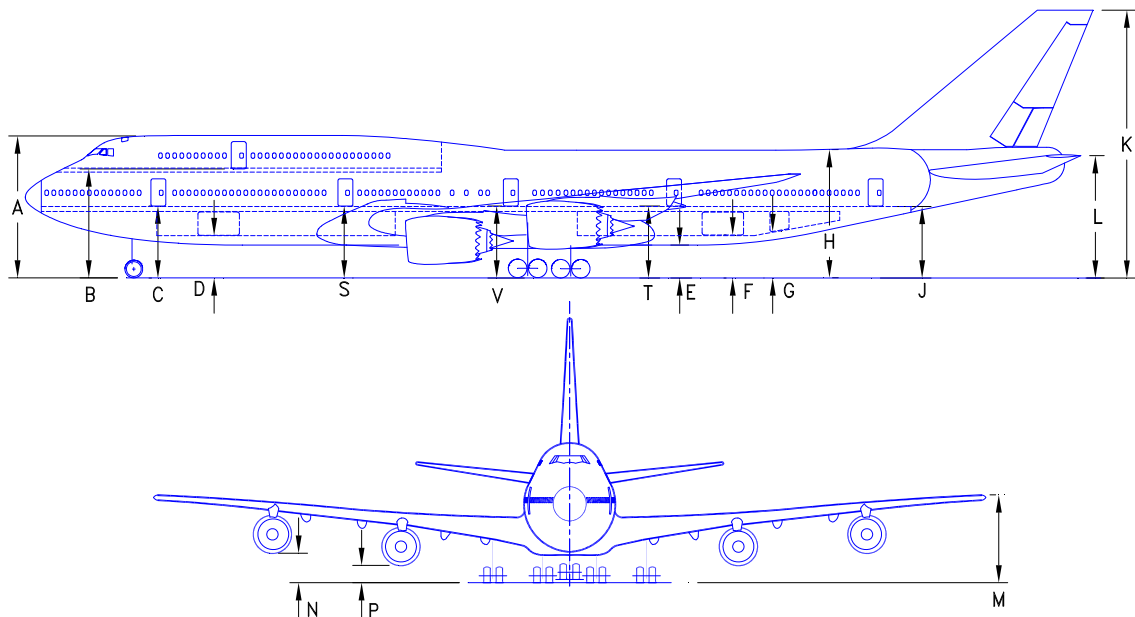
NOTES: VERTICAL CLEARANCES SHOWN OCCUR DURING MAXIMUM VARIATIONS OF AIRPLANE ATTITUDE. COMBINATIONS OF AIRPLANE LOADING/UNLOADING ACTIVITIES THAT PRODUCE THE GREATEST POSSIBLE VARIATIONS OF ATTITUDE WERE USED TO ESTABLISH THE VARIATIONS SHOWN. DURING ROUTINE SERVICING, THE AIRPLANE REMAINS RELATIVELY STABLE; PITCH AND ELEVATION CHANGES OCCUR SLOWLY.

AT MAJOR TERMINALS, A GSE TETHERING DEVICE MAY BE USED TO MAINTAIN STABILITY BETWEEN THE MAIN DECK DOOR SILL AND THE LOADING DOCK. CARGO BRIDGE ATTACHMENT FITTINGS LOCATED ON THE NOSE DOOR SILL AT THE FORWARD EDGE OF THE MAIN CARGO DOOR DECK MAY BE USED FOR NOSE DOOR SILL STABILIZATION.

2.3.1 GROUND CLEARANCES

MODEL 747-8F

PRELIMINARY

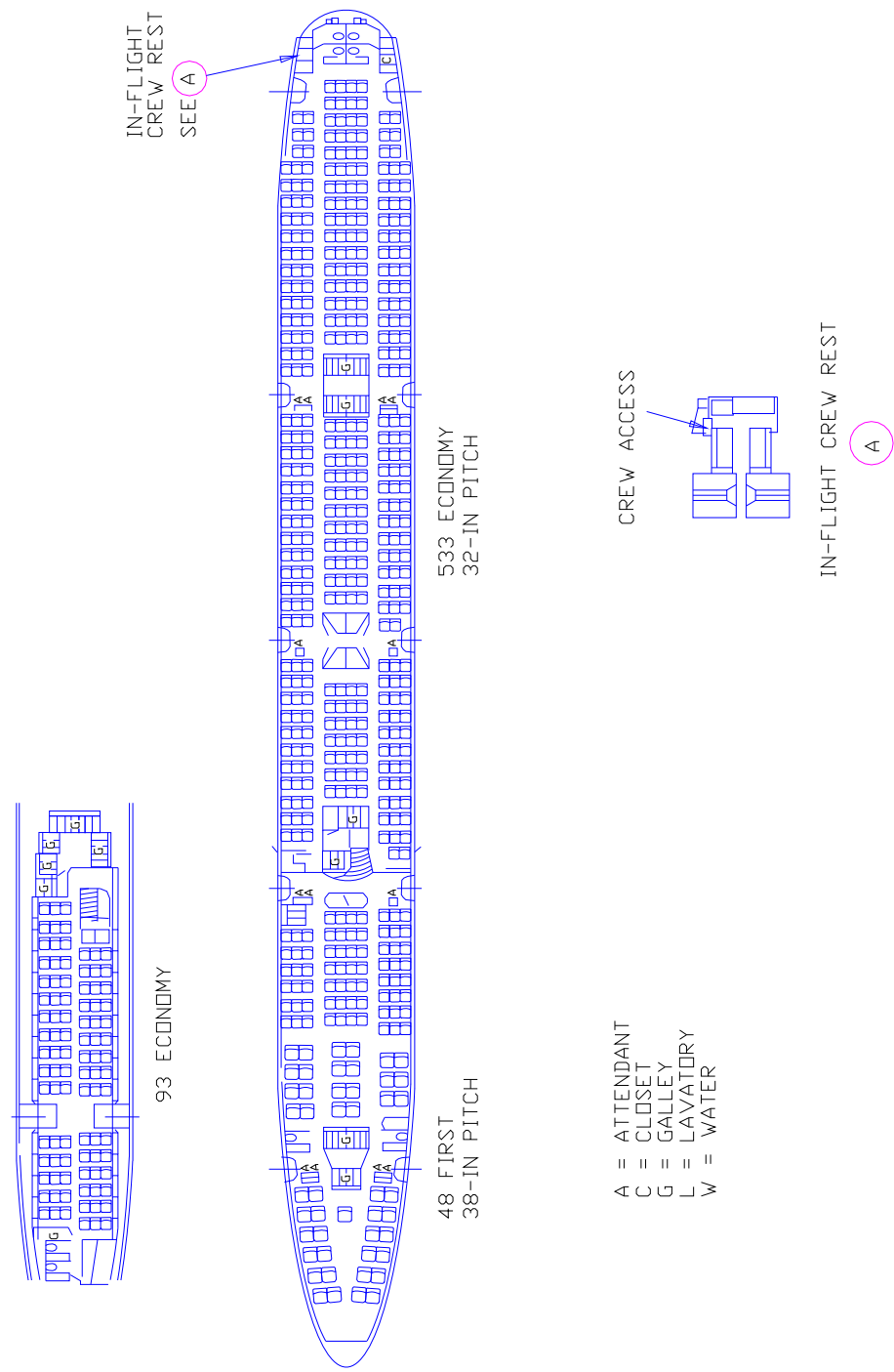


	MINIMUM		MAXIMUM	
	FT - IN	M	FT - IN	M
A	31 - 0	9.45	32 - 3	9.83
B	24 - 10	7.57	25 - 11	7.90
C	15 - 9	4.80	16 - 11	5.16
D	9 - 1	2.77	10 - 2	3.10
E	7 - 1	2.16	7 - 11	2.41
F	9 - 6	2.74	10 - 5	3.18
G	10 - 2	3.10	11 - 2	3.40
H	30 - 6	9.30	32 - 5	9.88
J	16 - 3	4.95	17 - 5	5.31
K	61 - 4	18.70	62 - 10	19.15
L	27 - 4	8.33	28 - 10	8.79
M	18 - 7	5.66	20 - 4	6.20
N	5 - 0	1.53	6 - 0	1.83
P	2 - 2	0.66	3 - 0	0.91
S	16 - 0	4.88	16 - 10	5.13
T	16 - 3	4.95	17 - 0	5.18
V	16 - 3	4.95	16 - 9	5.11

NOTES: VERTICAL CLEARANCES SHOWN OCCUR DURING MAXIMUM VARIATIONS OF AIRPLANE ATTITUDE. COMBINATIONS OF AIRPLANE LOADING/UNLOADING ACTIVITIES THAT PRODUCE THE GREATEST POSSIBLE VARIATIONS OF ATTITUDE WERE USED TO ESTABLISH THE VARIATIONS SHOWN. DURING ROUTINE SERVICING, THE AIRPLANE REMAINS RELATIVELY STABLE; PITCH AND ELEVATION CHANGES OCCUR SLOWLY.

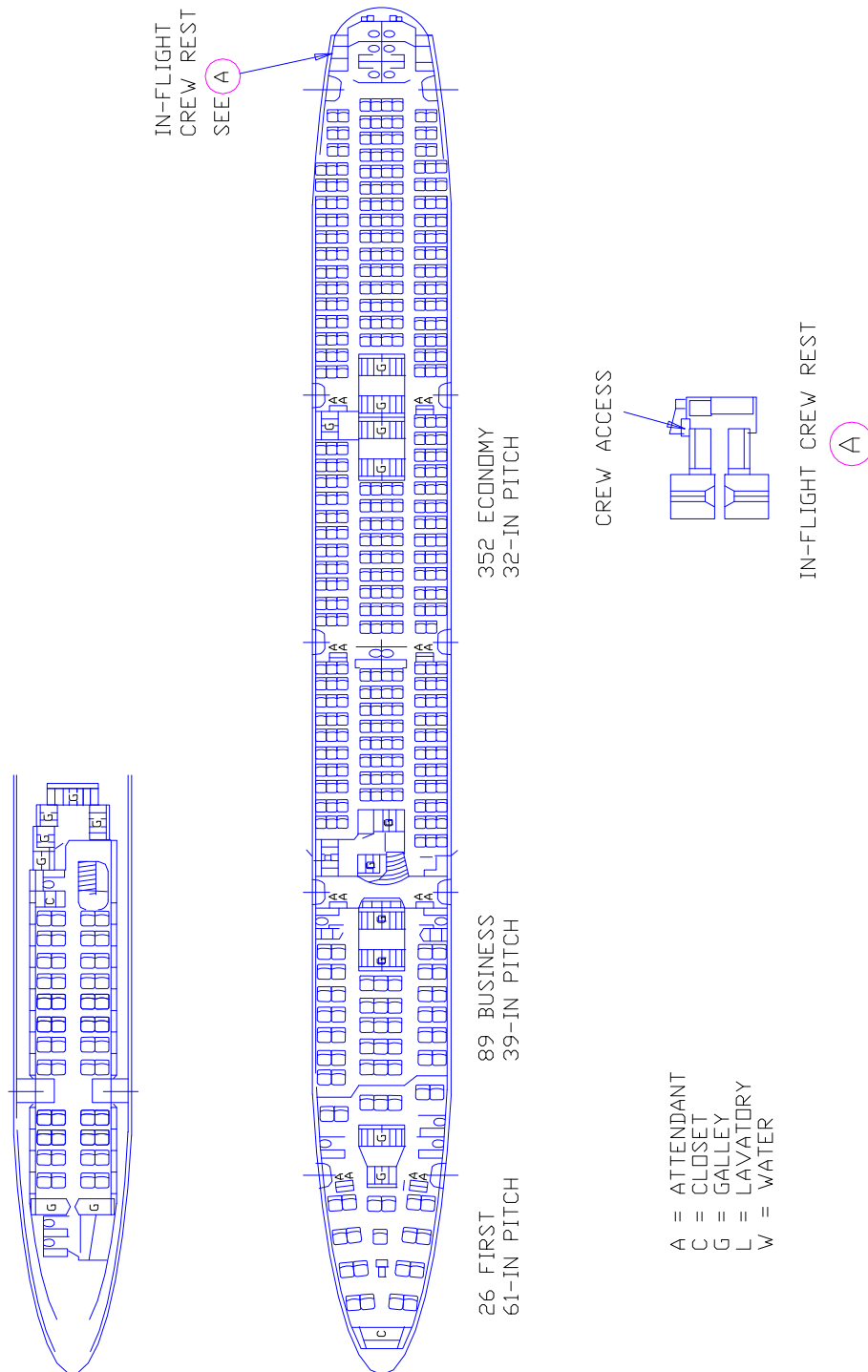
2.3.2 GROUND CLEARANCES MODEL 747-8

PRELIMINARY



2.4.1 TYPICAL INTERIOR ARRANGEMENTS, TWO CLASS, 581 PASSENGERS
MODEL 747-8

PRELIMINARY



2.4.2 TYPICAL INTERIOR ARRANGEMENTS, THREE CLASS, 467 PASSENGERS
MODEL 747-8

PRELIMINARY

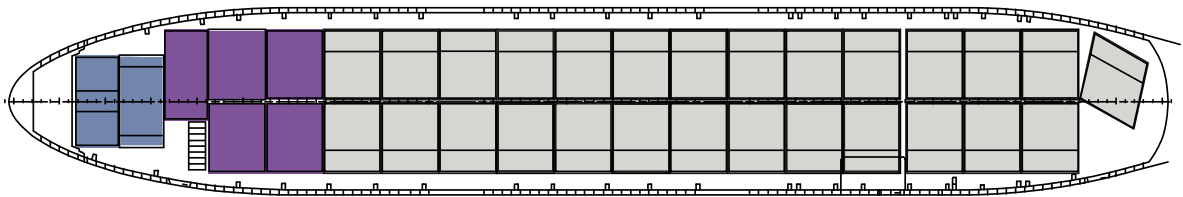
Main deck

■ (2) 96-in x 125-in x 8-ft contoured pallets, $15.3 \text{ m}^3 + 17.2 \text{ m}^3$ (540 + 607 ft³) = 32.5 m^3 (1,147 ft³)

■ (5) 96-in x 125-in x 8-ft pallets, 17.4 m^3 (613 ft³) = 86.8 m^3 (3,065 ft³)

■ (27) 96-in x 125-in contoured pallets, 21.2 m^3 (750 ft³) = 573.4 m^3 (20,250 ft³)

Total 692.7 m^3 (24,462 ft³)



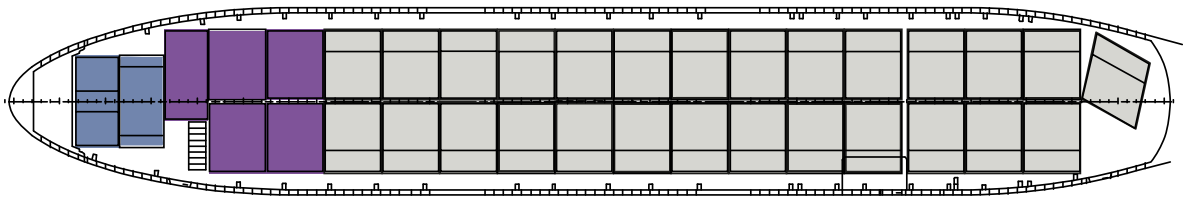
Main deck (alternate)

■ (2) 88-in x 125-in x 8-ft contoured pallets, $14.0 \text{ m}^3 + 15.7 \text{ m}^3$ (493 + 554 ft³) = 29.6 m^3 (1,047 ft³)

■ (5) 88-in x 125-in x 8-ft pallets, 15.9 m^3 (560 ft³) = 79.3 m^3 (2,800 ft³)

■ (27) 88-in x 125-in contoured pallets, 19.3 m^3 (680 ft³) = 519.9 m^3 (18,360 ft³)

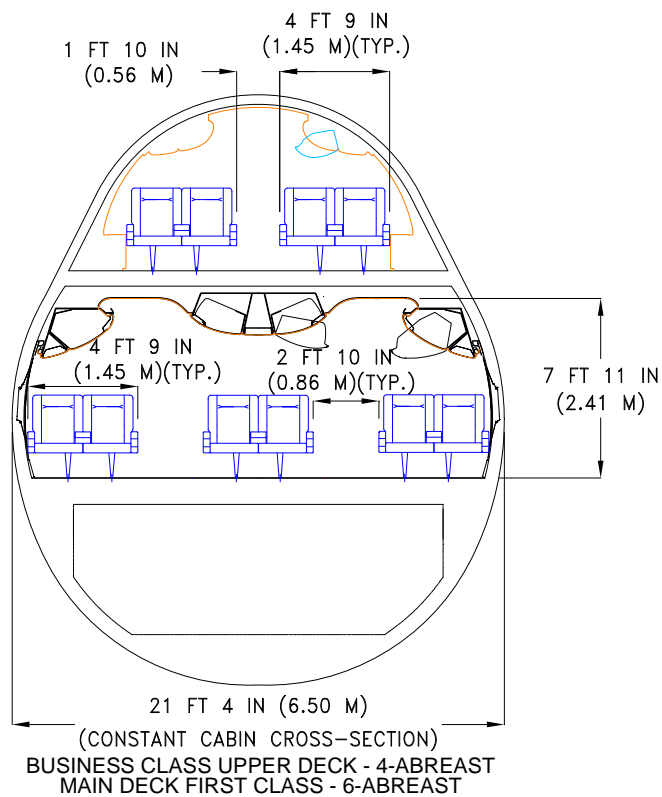
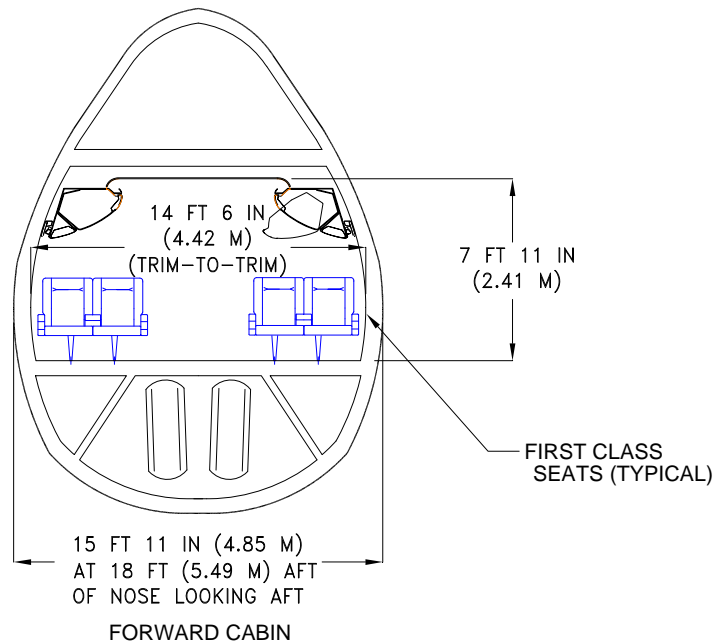
Total 628.8 m^3 (22,207 ft³)



2.4.3 TYPICAL INTERIOR ARRANGEMENTS

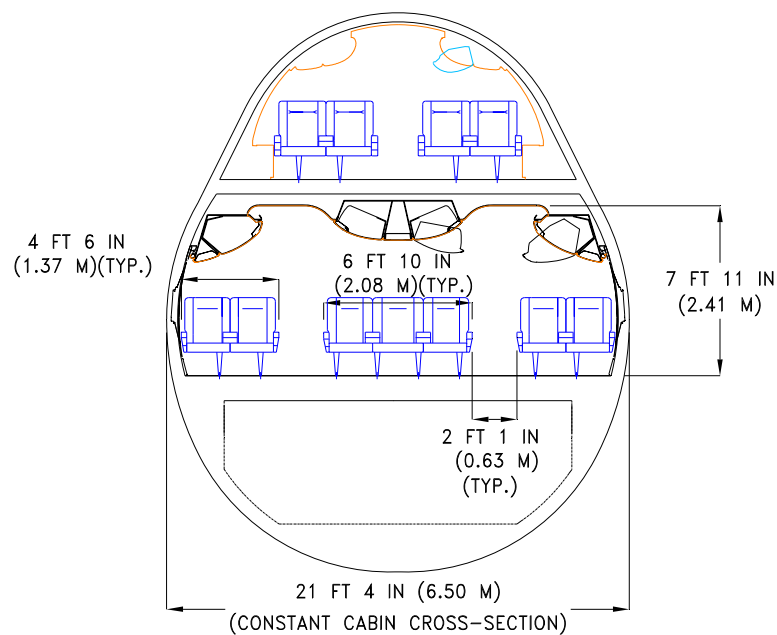
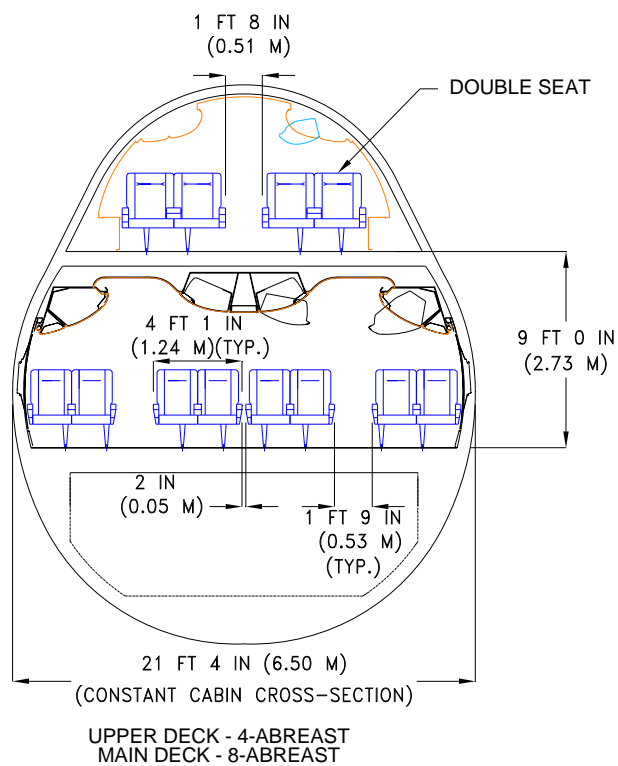
MODEL 747-8F

PRELIMINARY



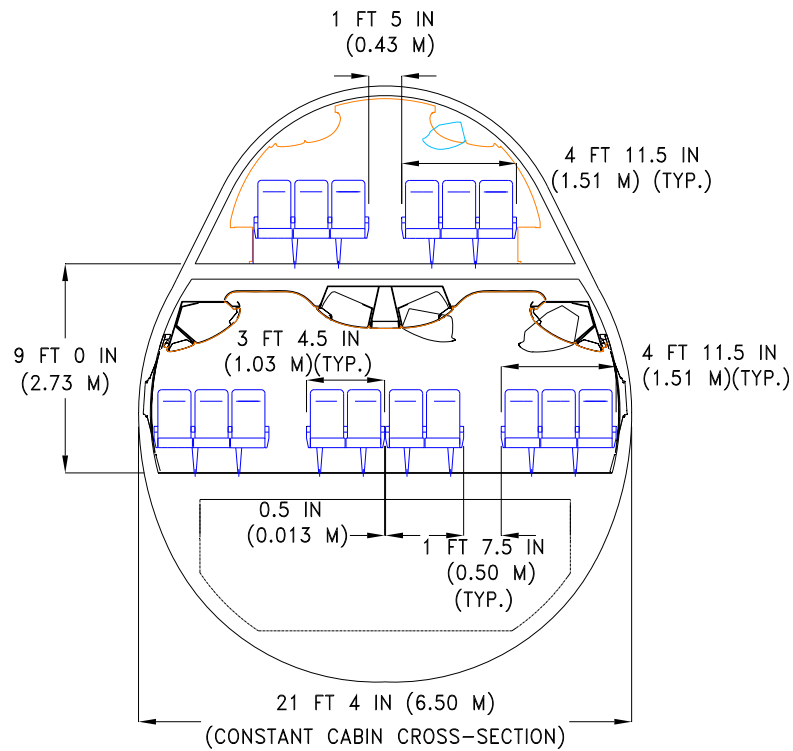
2.5.1 CABIN CROSS-SECTION MODEL 747-8

PRELIMINARY

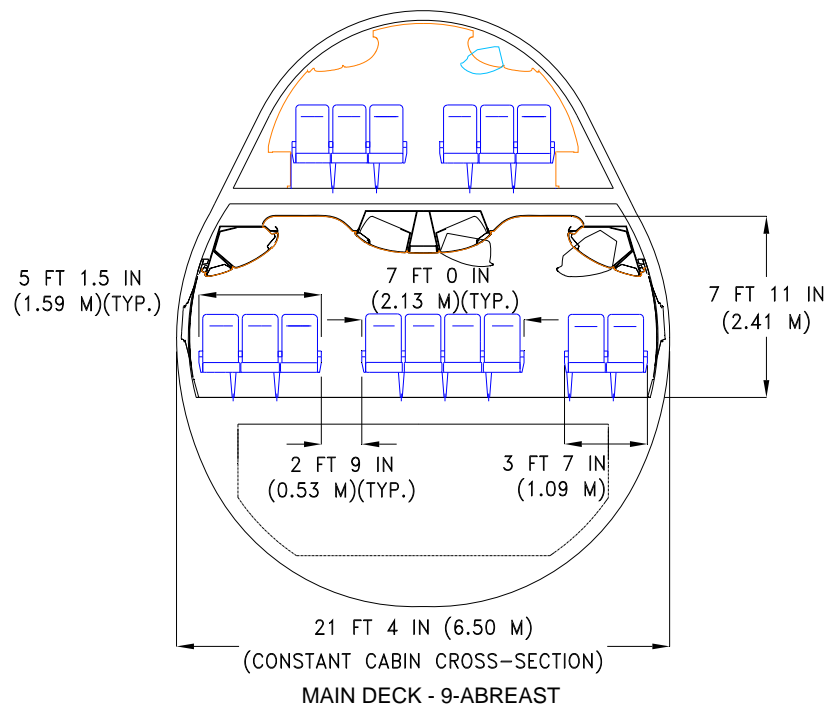


2.5.2 CABIN CROSS-SECTION MODEL 747-8

PRELIMINARY

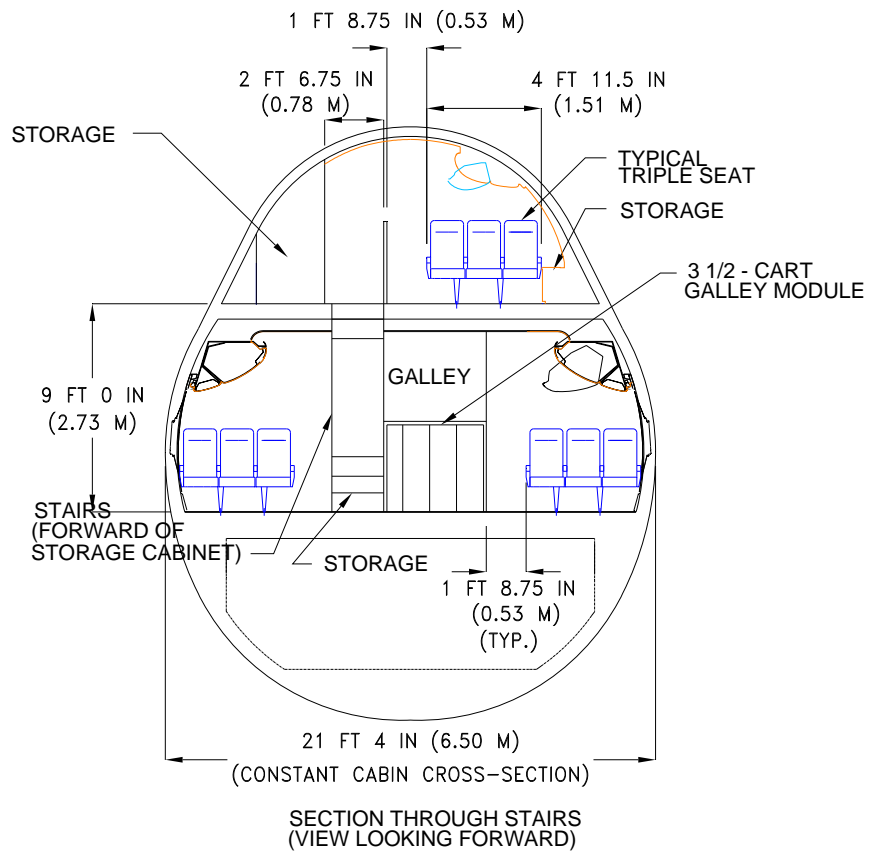


UPPER DECK - 6-ABREAST
MAIN DECK - 10-ABREAST



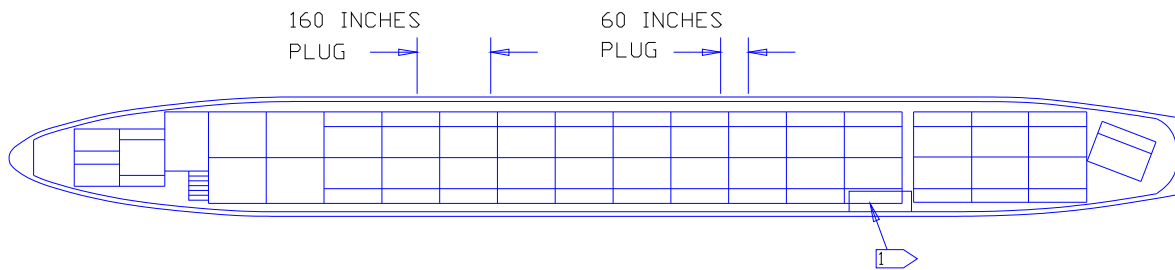
2.5.3 CABIN CROSS-SECTION MODEL 747-8

PRELIMINARY

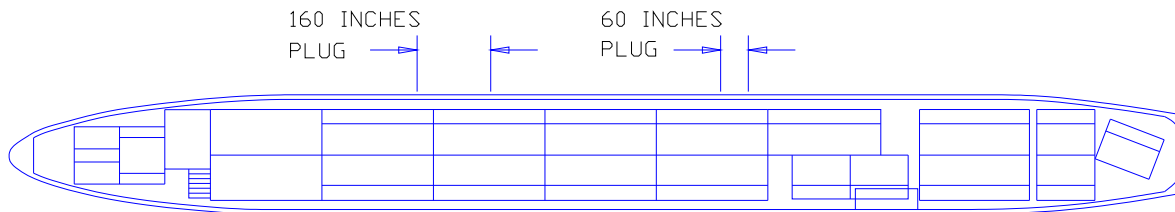


2.5.4 CABIN CROSS-SECTION MODEL 747-8

PRELIMINARY

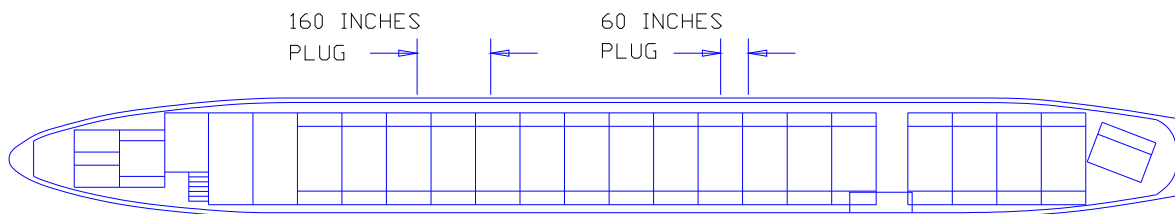


⟨34⟩ 96 INCHES X 125 INCHES PALLETS



⟨13⟩ 96 INCHES X 238.5 INCHES PALLETS

⟨8⟩ 96 INCHES X 125 INCHES PALLETS



⟨19⟩ 96 INCHES X 196 INCHES PALLETS

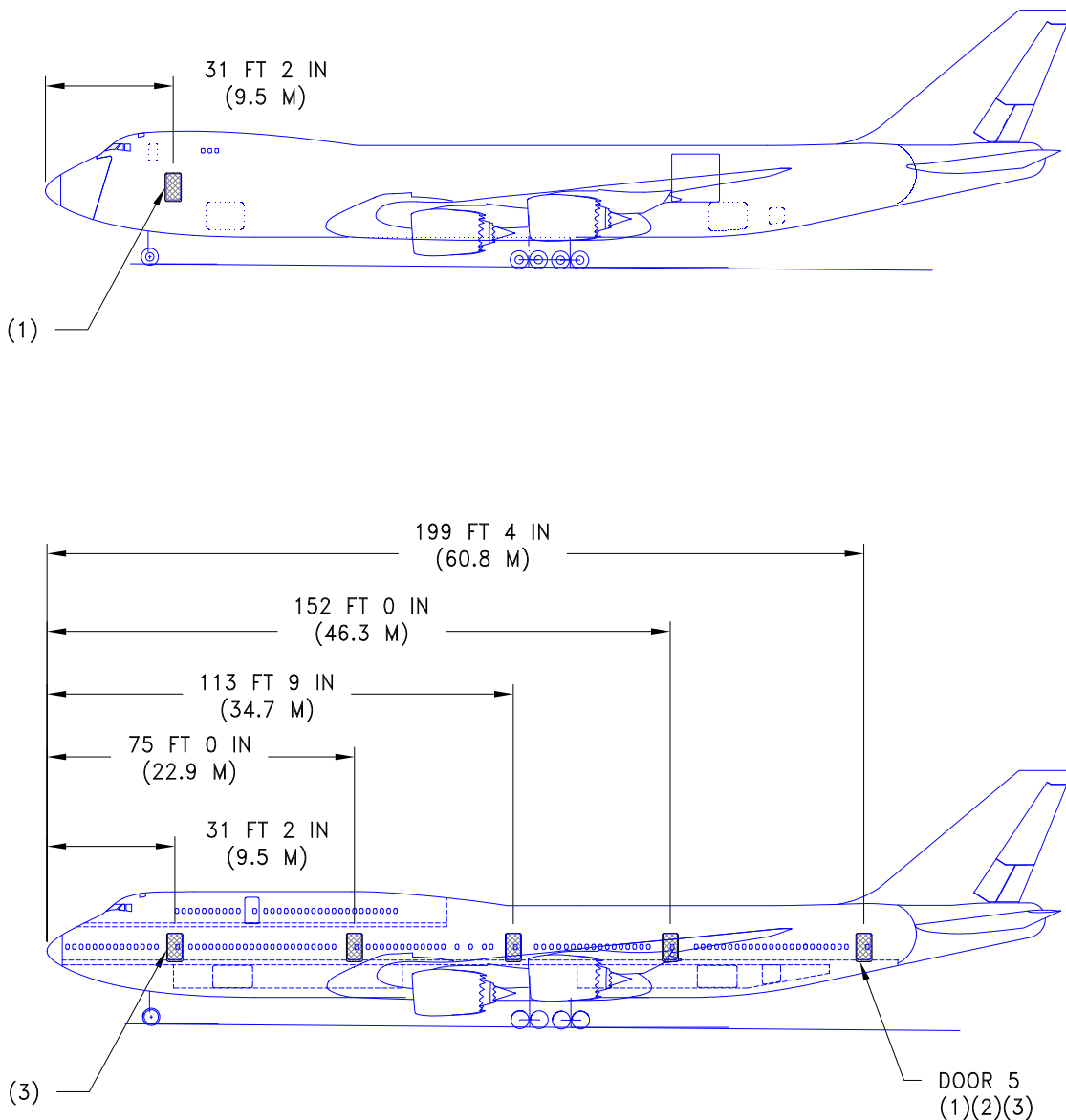
⟨4⟩ 96 INCHES X 125 INCHES PALLETS

⟨24 ADDITIONAL PALLET LOCKS REQUIRED⟩

1 88 INCHES X 125 INCHES PALLETS ARE LIMITED TO THESE POSITIONS ONLY.

2.6 LOWER CARGO COMPARTMENTS - CONTAINERS AND BULK CARGO MODEL 747-8F

PRELIMINARY

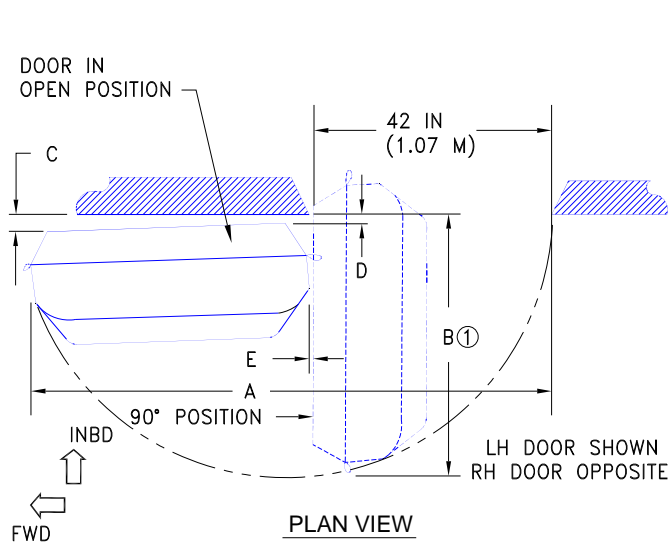


NOTES:

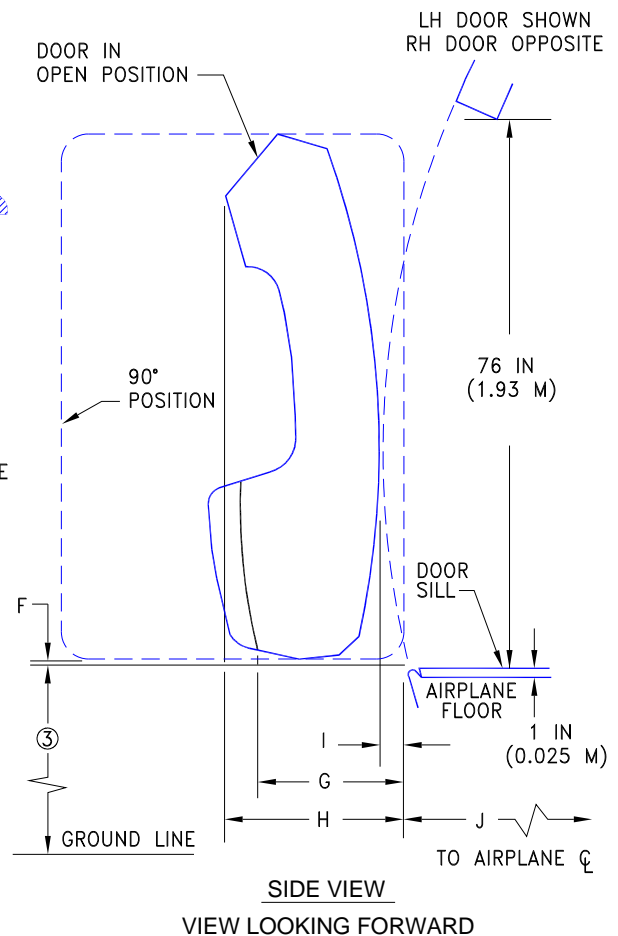
- (1) 1 PASSENGER DOOR – LEFT SIDE ONLY FOR THE 747-8 FREIGHTER
10 PASSENGER DOORS – 5 EACH SIDE FOR THE 747-8 INTERCONTINENTAL
DOOR OPENING SIZE = 42 BY 76 IN (1.07 BY 1.93 M)
OVERALL DOOR SIZE = 47 BY 76 IN (1.19 BY 1.93 M)
- (2) SEE SECTION 2.3 FOR DOOR SILL HEIGHTS
- (3) LH DOOR NO. 1 AND NO. 5 ON 747-8 INTERCONTINENTAL

2.7.1 DOOR CLEARANCES - MAIN ENTRY DOOR LOCATIONS MODEL 747-8, 747-8F

PRELIMINARY



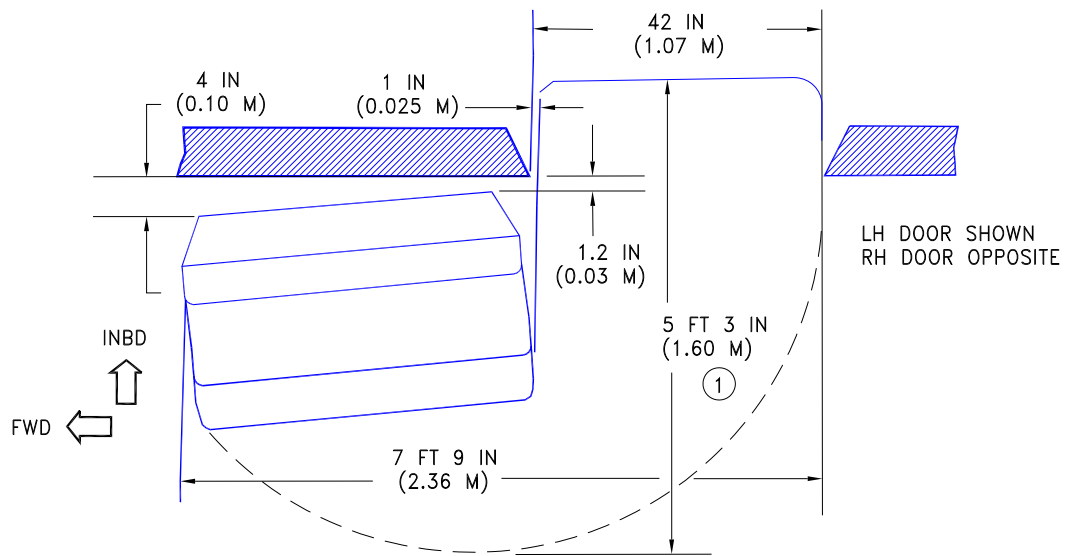
	DOOR NUMBER			
	1 ②	2	3	4
A	7 FT 6 IN	7 FT 6 IN	7 FT 6 IN	7 FT 6 IN
	2.29 M	2.29 M	2.29 M	2.29 M
B ①	3 FT 9 IN	4 FT 0 IN	3 FT 8 IN	3 FT 8 IN
	1.14 M	1.2 M	1.1 M	1.1 M
C	6.7 IN	5 IN	4 IN	4 IN
	0.17 M	0.127 M	0.1 M	0.1 M
D	1 IN	1 IN	1 IN	1 IN
	0.025 M	0.025 M	0.025 M	0.025 M
E	1 IN	1 IN	1 IN	1 IN
	0.025 M	0.025 M	0.025 M	0.025 M
F	2 IN	2 IN	1 IN	1 IN
	0.05 M	0.05 M	0.025 M	0.025 M
G ①	1 FT 7 IN	1 FT 7 IN	1 FT 10 IN	1 FT 10 IN
	0.48 M	0.48 M	0.56 M	0.56 M
H ①	1 FT 11 IN	1 FT 11 IN	2 FT 0 IN	2 FT 0 IN
	0.58 M	0.58 M	0.61 M	0.61 M
I ①	1 IN	3 IN	0	3 IN
	0.025 M	0.076 M	0	0.076 M
J ①	9 FT 6 IN	10 FT 5 IN	10 FT 8 IN	10 FT 5 IN
	2.90 M	3.18 M	3.25 M	3.18 M



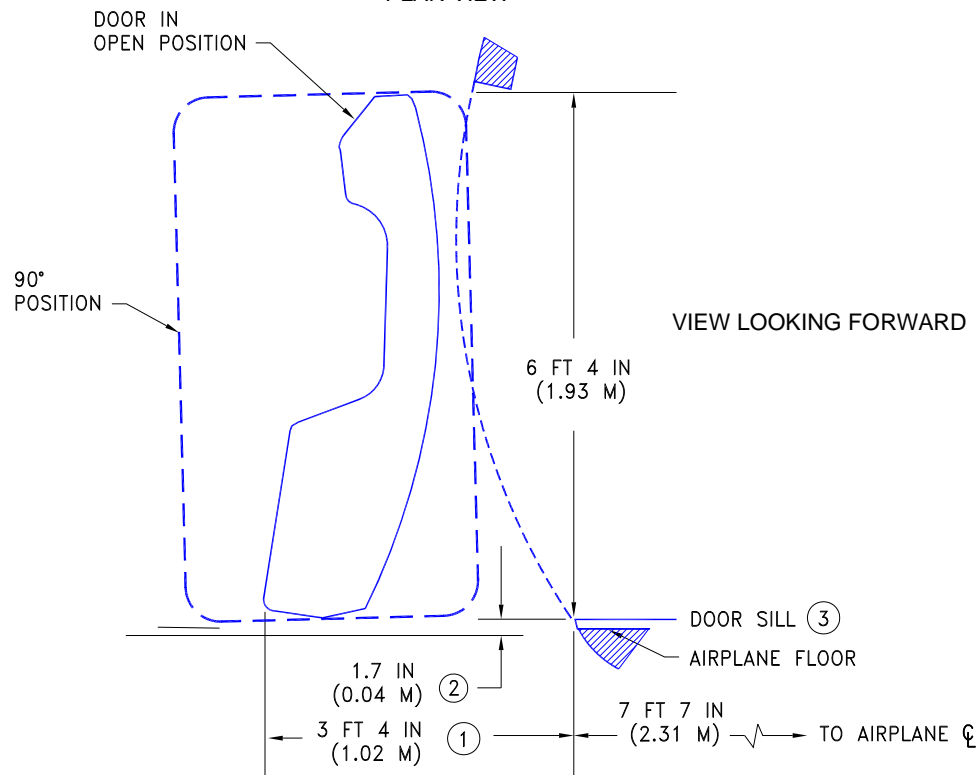
- ① MEASURED AT DOOR OPENING CENTERLINE AT DOOR SILL LEVEL AT 90° FROM AIRPLANE CENTERLINE.
- ② LH SIDE ONLY ON 747-8F.
- ② SEE SEC. 2.3 FOR DOOR SILL HEIGHTS

2.7.2 DOOR CLEARANCES - MAIN ENTRY DOORS 1-4 MODEL 747-8, 747-8F

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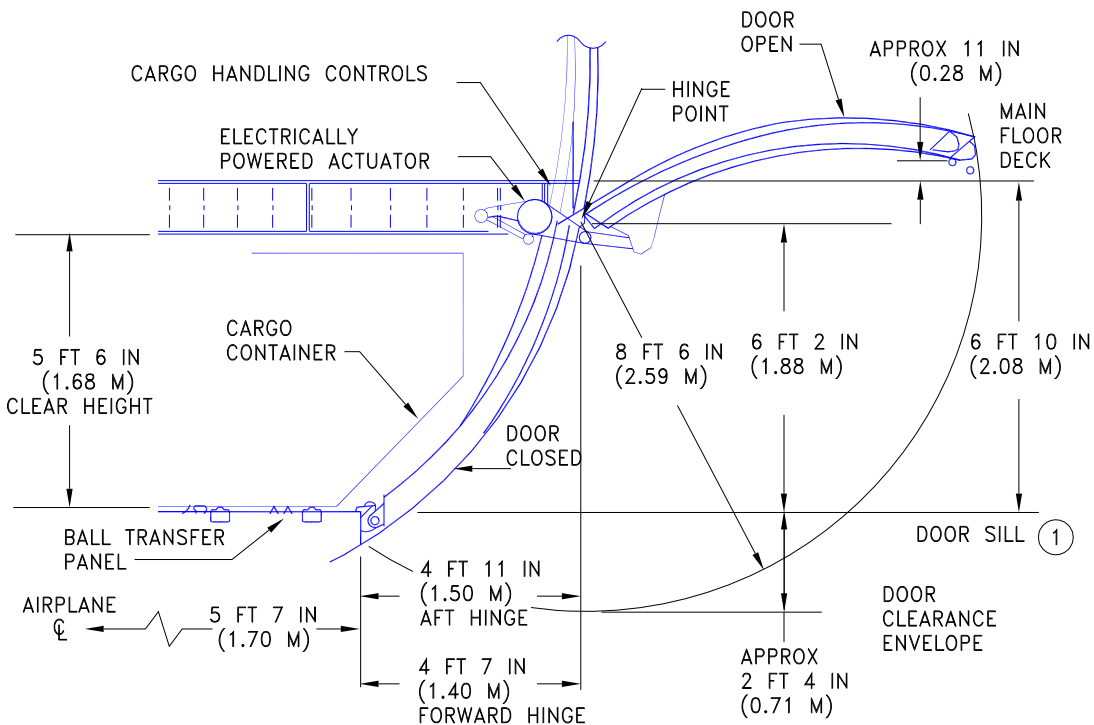
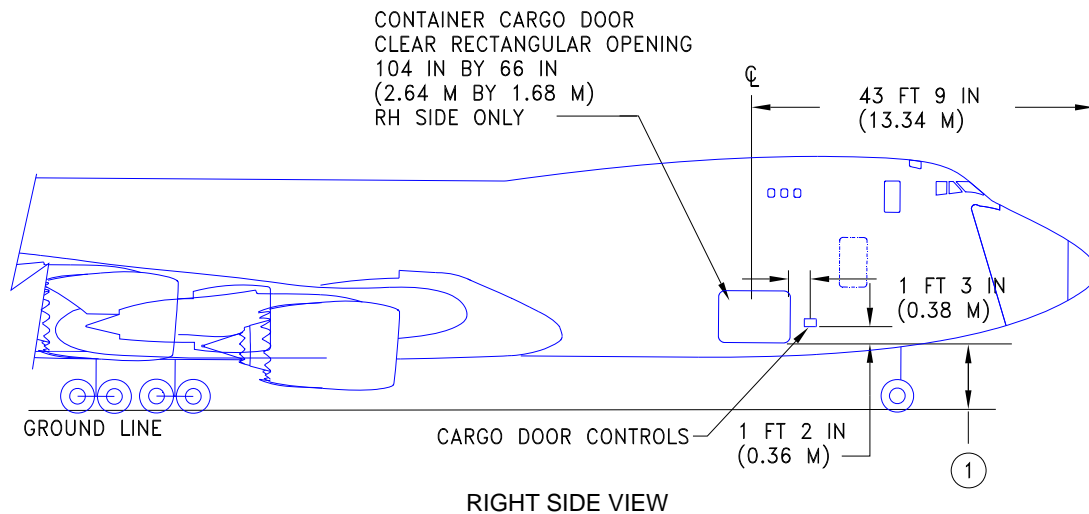
PLAN VIEW



- ① MEASURED AT DOOR OPENING CENTERLINE AT DOOR SILL LEVEL AT 90° FROM AIRPLANE CENTERLINE
- ② DOOR HINGE IS INCLINED 3 DEGREES FROM VERTICAL
- ③ SEE SEC. 2.3 FOR DOOR SILL HEIGHTS
- ④ LH SIDE ONLY ON 747-8F

2.7.3 DOOR CLEARANCES - MAIN ENTRY DOOR 5 MODEL 747-8, 747-8F

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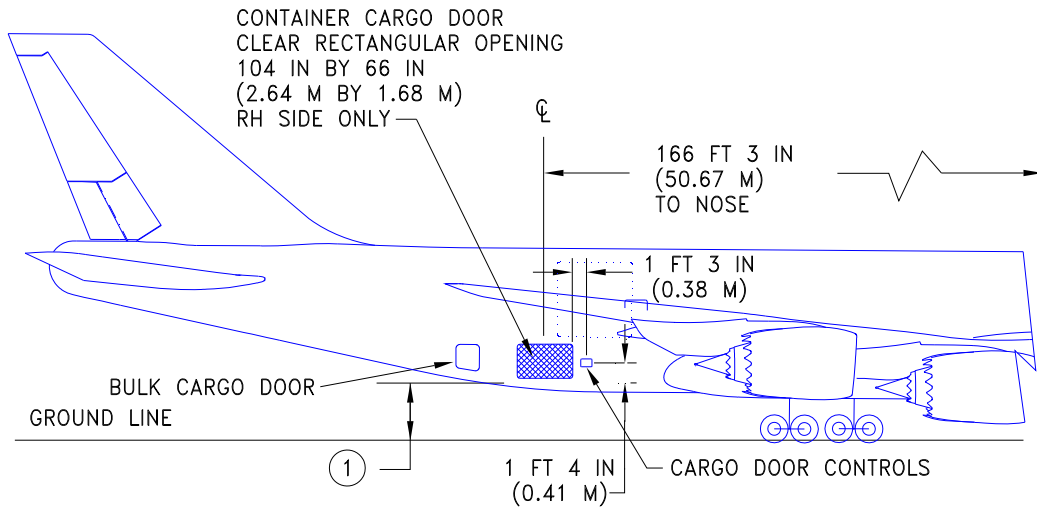


CONTAINER CARGO DOOR - VIEW LOOKING FORWARD

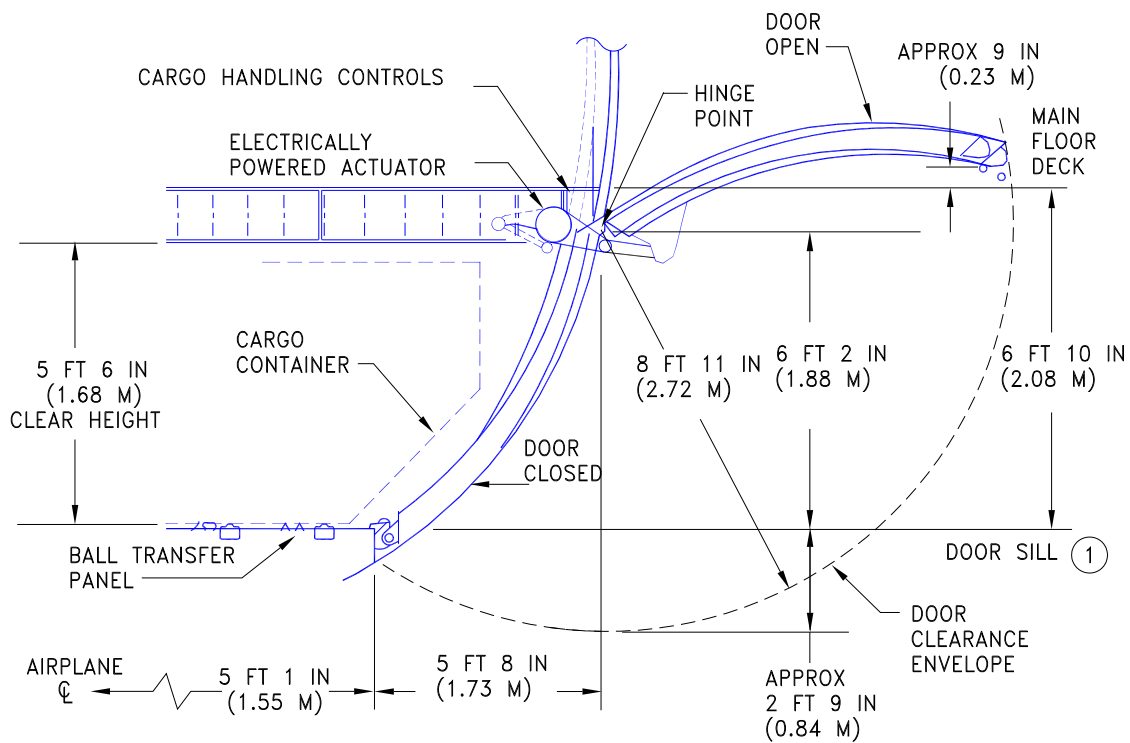
① SEE SECTION 2.3 FOR DOOR SILL HEIGHTS

2.7.4 DOOR CLEARANCES – LOWER FORWARD CARGO COMPARTMENT MODEL 747-8, 747-8F

PRELIMINARY



RIGHT SIDE VIEW

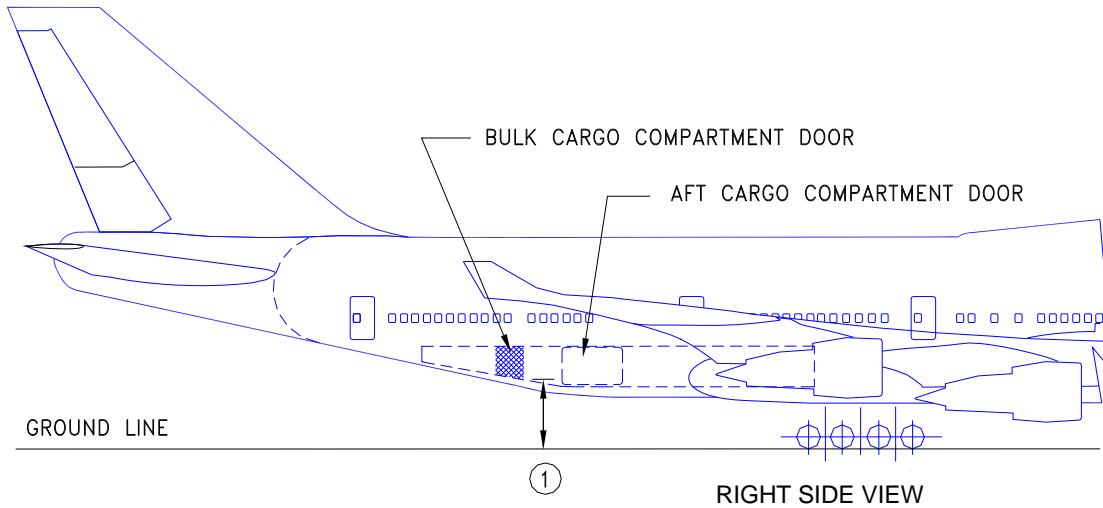


CONTAINER CARGO DOOR - VIEW LOOKING FORWARD

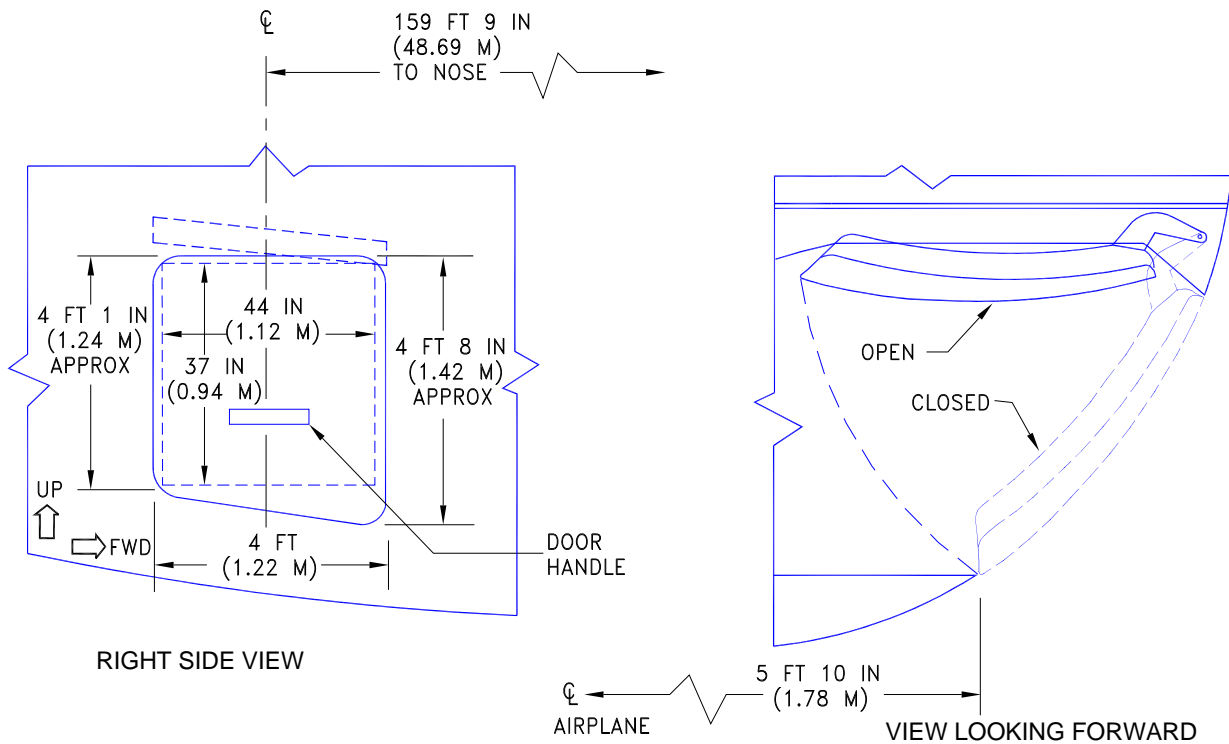
① SEE SECTION 2.3 FOR DOOR SILL HEIGHTS

2.7.5 DOOR CLEARANCES – LOWER AFT CARGO COMPARTMENT MODEL 747-8, 747-8F

PRELIMINARY

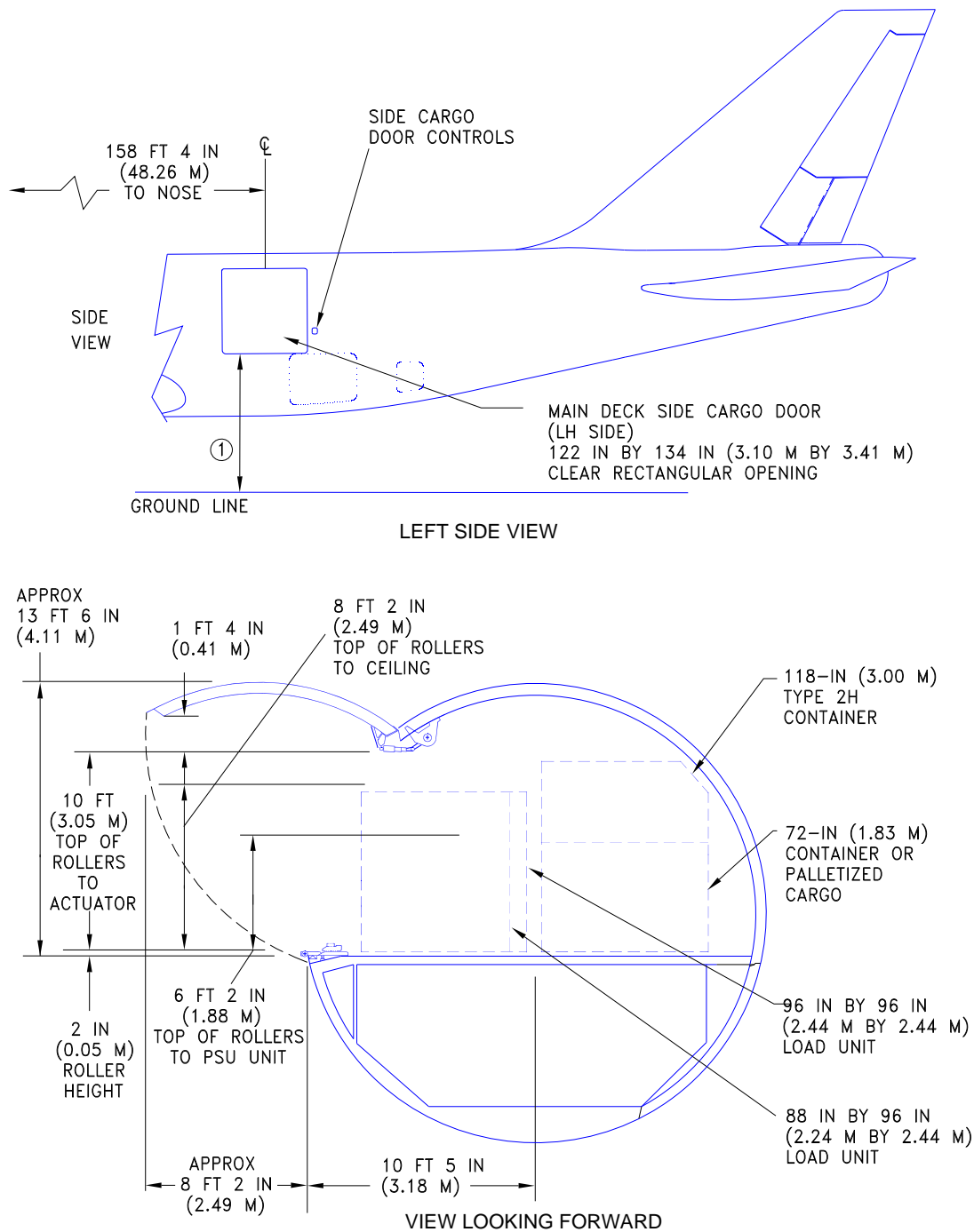


① SEE SECTION 2.3 FOR DOOR SILL HEIGHTS



2.7.6 DOOR CLEARANCES - BULK CARGO COMPARTMENT MODEL 747-8

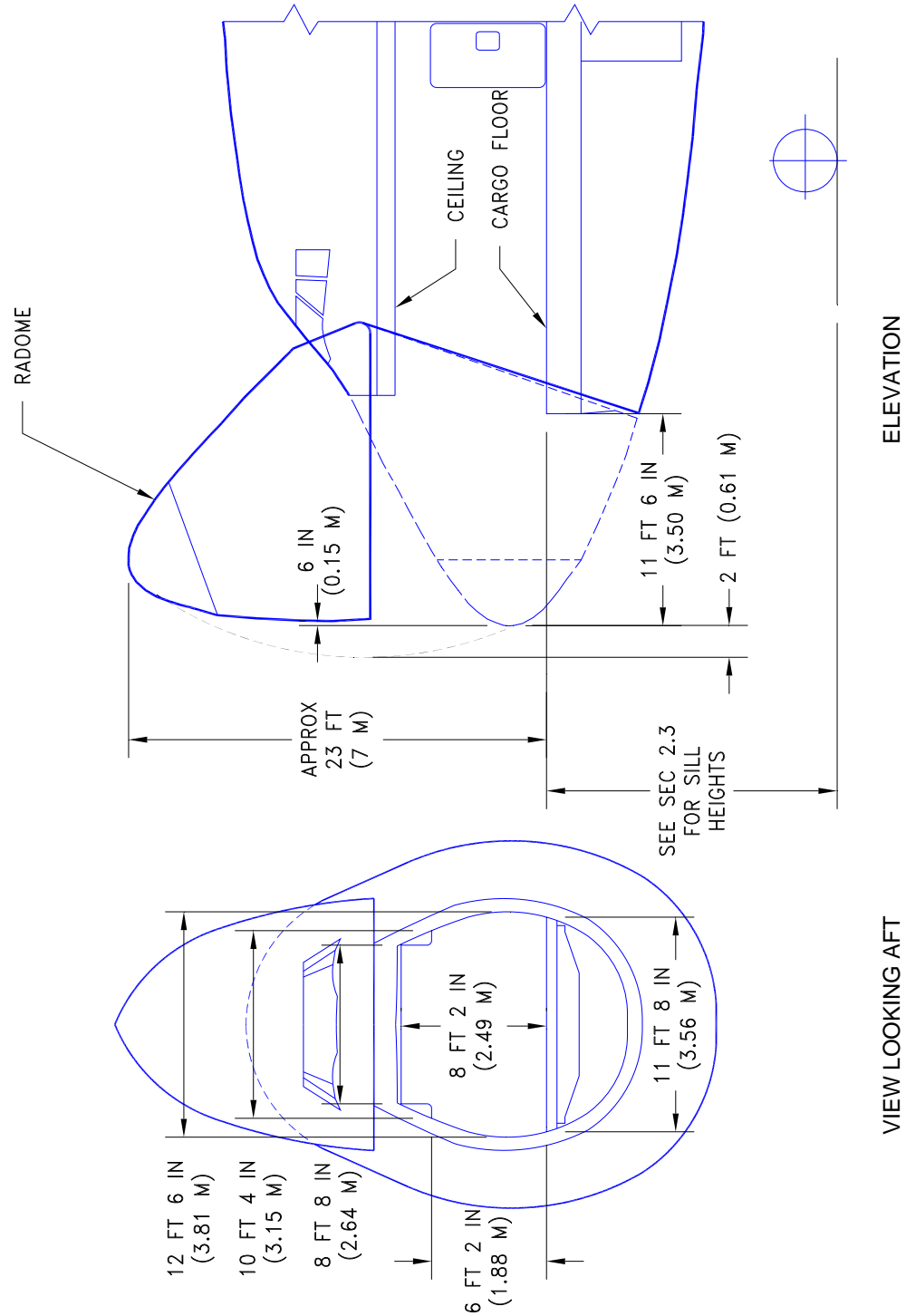
PRELIMINARY



① SEE SECTION 2.3 FOR DOOR SILL HEIGHTS

2.7.7 DOOR CLEARANCES – MAIN DECK CARGO DOOR MODEL 747-8F

PRELIMINARY



2.7.8 DOOR CLEARANCES - NOSE CARGO DOOR MODEL 747-8F

PRELIMINARY

3.0 AIRPLANE PERFORMANCE

3.1 General Information

3.2 Payload/Range for 0.845 Mach Cruise

3.3 F.A.R. Takeoff Runway Length Requirements

3.4 F.A.R. Landing Runway Length Requirements

PRELIMINARY

3.0 AIRPLANE PERFORMANCE

3.1 General Information

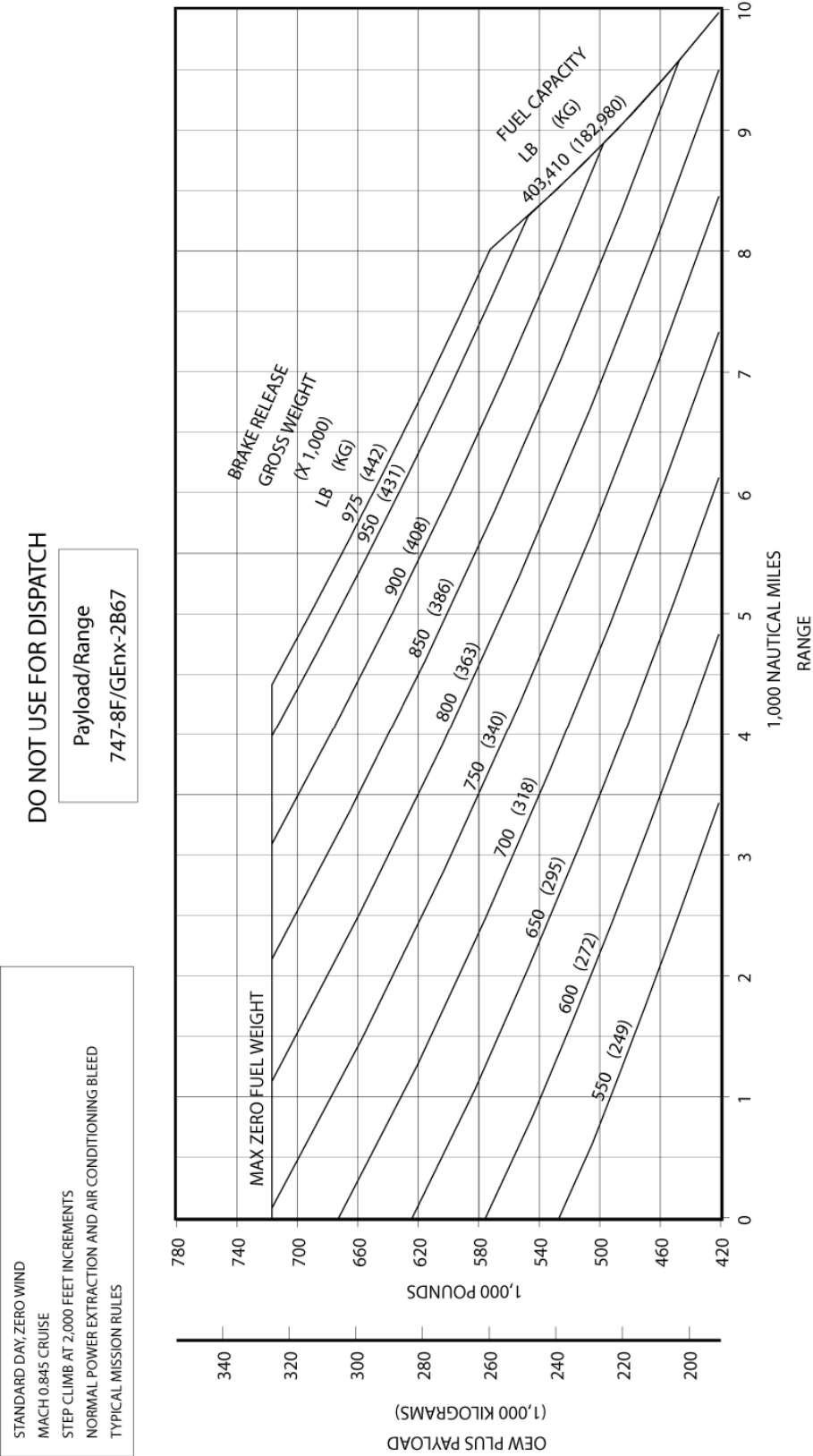
The graphs in Section 3.2 provide information on operational empty weight (OEW) and payload, trip range, brake release gross weight, and fuel limits for the GENx engine. To use these graphs, if the trip range and zero fuel weight (OEW + payload) are known, the approximate brake release weight can be found, limited by fuel quantity. Examples of loading conditions under certain OEW's are illustrated in each graph.

The graphs in Section 3.3 provide information on F.A.R. takeoff runway length requirements with the GENx engines at different pressure altitudes. Maximum takeoff weights shown on the graphs are the heaviest for the particular airplane models with the corresponding engines. Standard day temperatures for pressure altitudes shown on the F.A.R. takeoff graphs are given below:

PRESSURE ALTITUDE		STANDARD DAY TEMP	
FEET	METERS	°F	°C
0	0	59.0	15.00
2,000	610	51.9	11.04
4,000	1,219	44.7	7.06
6,000	1,829	37.6	3.11
8,000	2,438	30.5	-0.85
10,000	3,048	23.3	-4.81

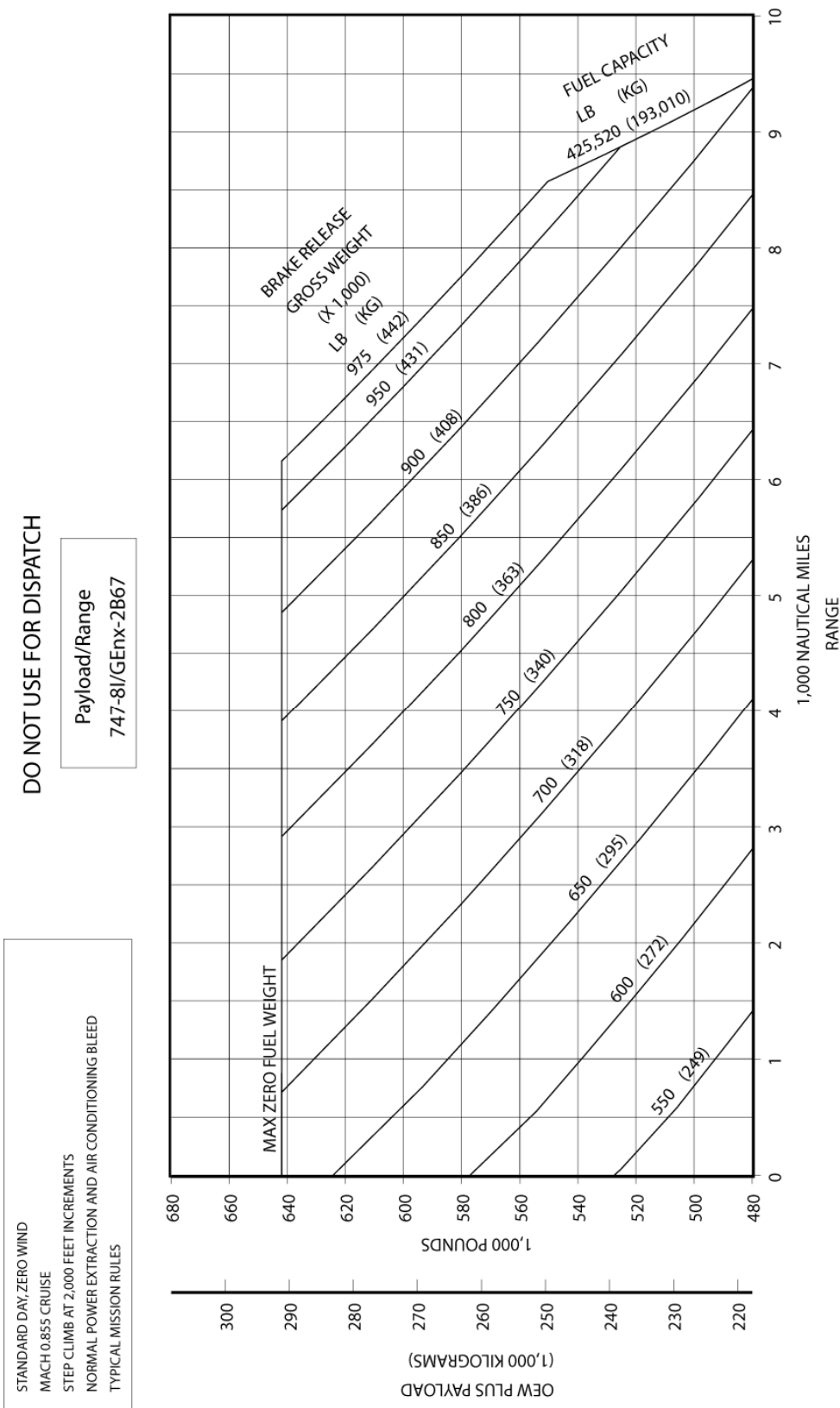
The graphs in Section 3.4 provide information on landing runway length requirements for different airplane weights and airport altitudes. The maximum landing weights shown are the heaviest for the particular airplane model.

PRELIMINARY



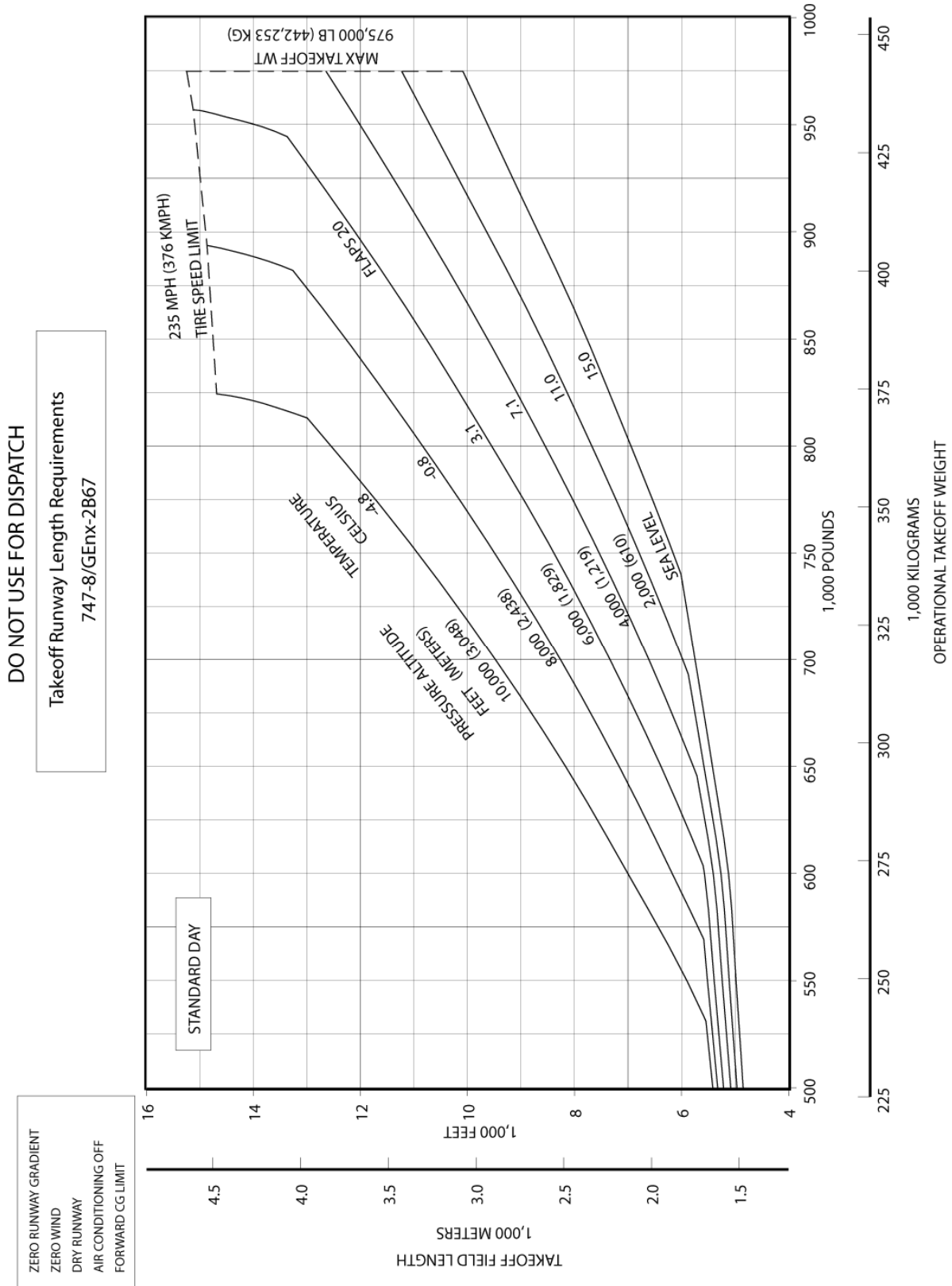
3.2.1 PAYLOAD/RANGE FOR 0.845 MACH CRUISE MODEL 747-8F

PRELIMINARY



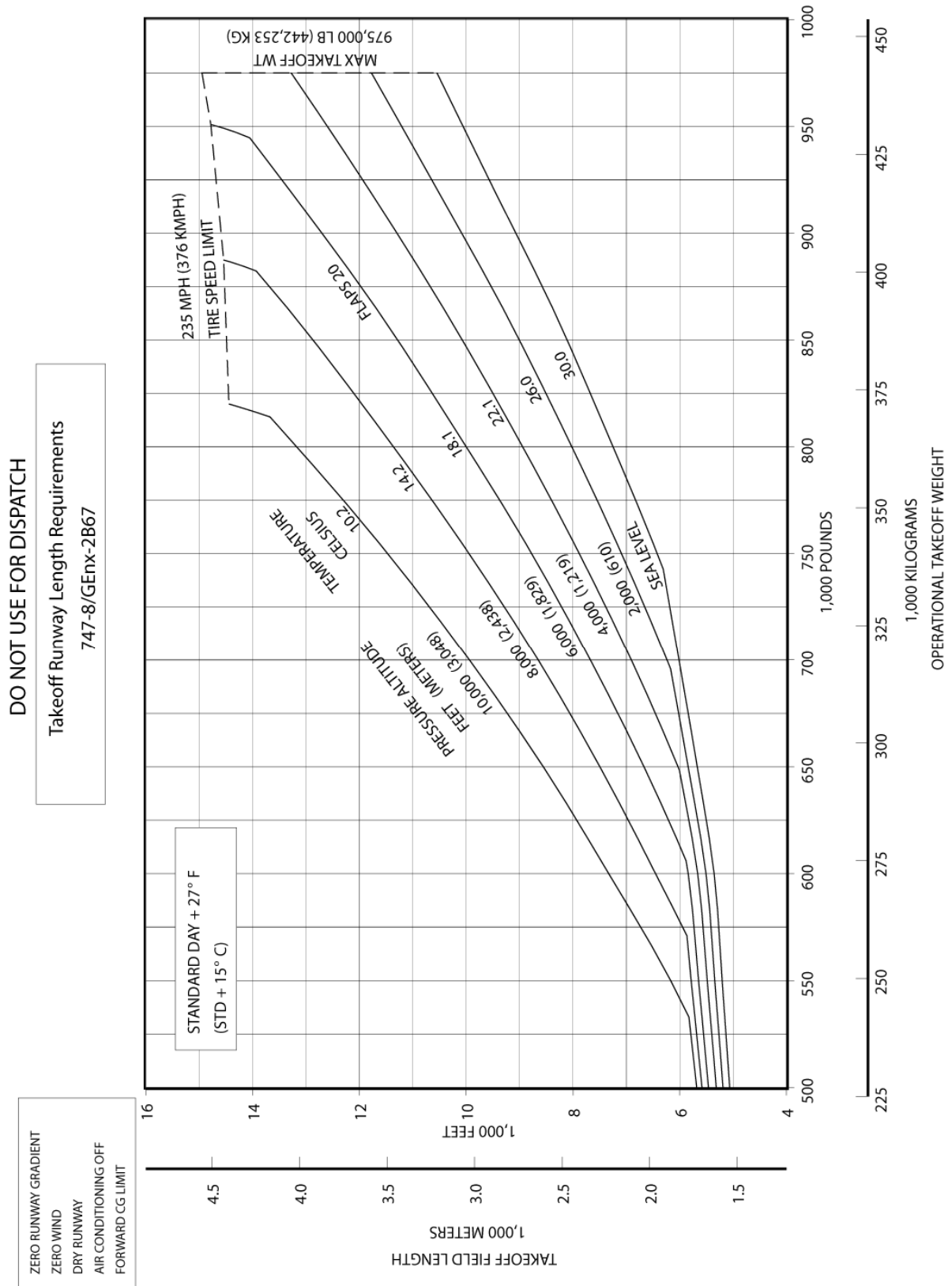
3.2.2 PAYLOAD/RANGE FOR 0.845 MACH CRUISE
MODEL 747-8I

PRELIMINARY



3.3.1 F.A.R. TAKEOFF RUNWAY LENGTH REQUIREMENTS - STANDARD DAY MODEL 747-8F AND 747-8I

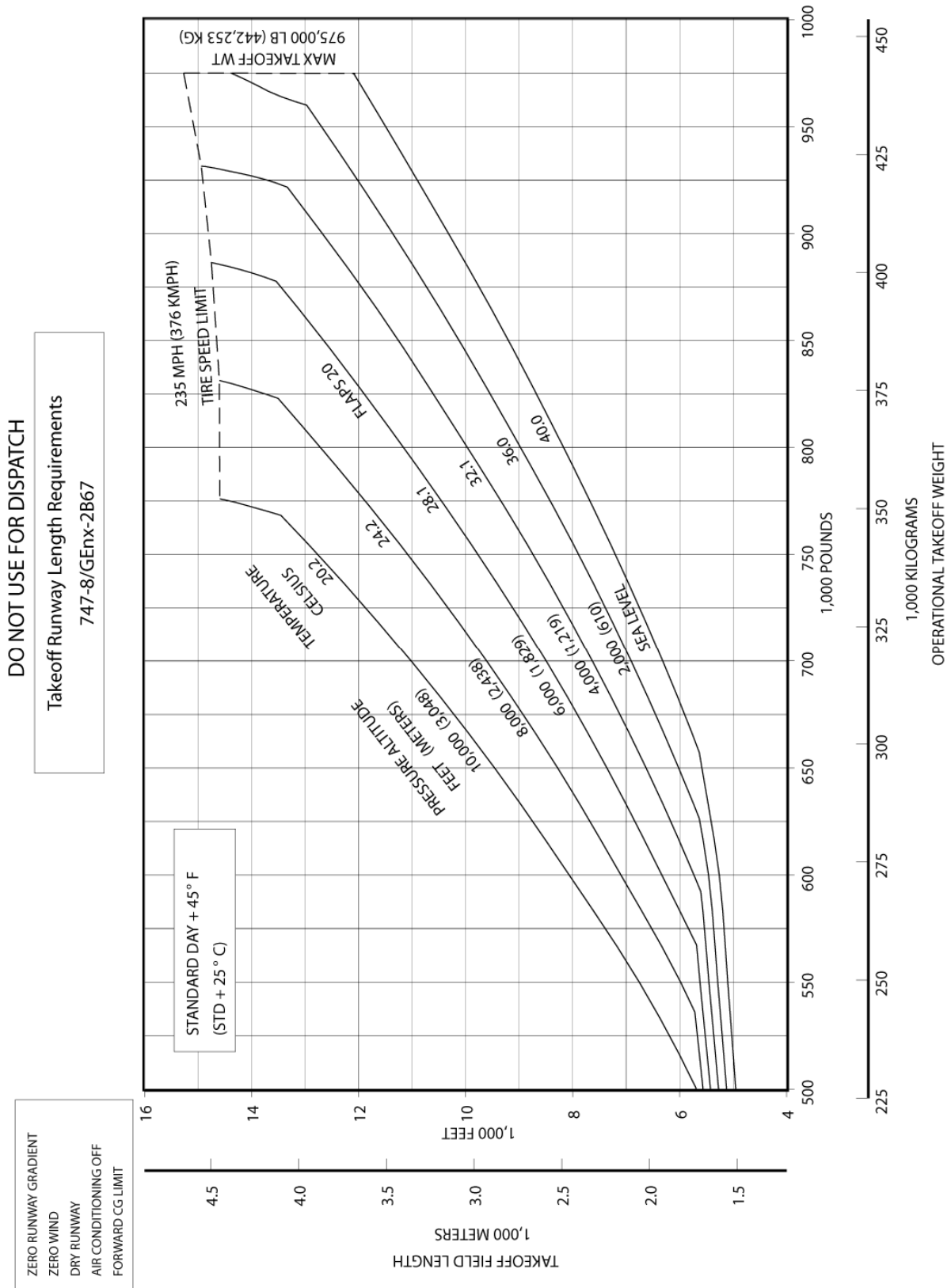
PRELIMINARY



3.3.2 F.A.R. TAKEOFF RUNWAY LENGTH REQUIREMENTS - STANDARD DAY + 27° F (STD + 15° C) MODEL 747-8F AND 747-8I

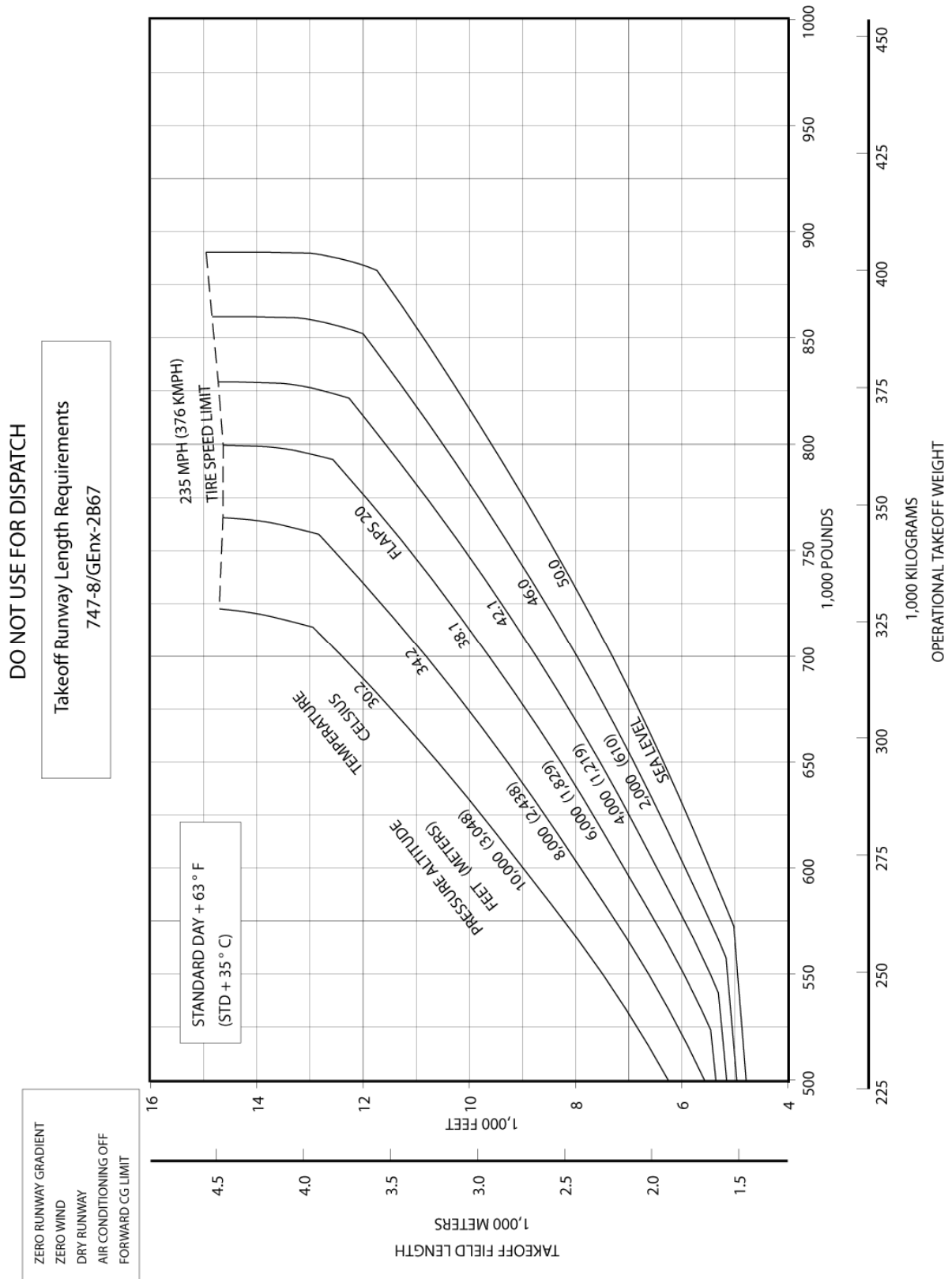
D6-58326-3

PRELIMINARY



3.3.3 F.A.R. TAKEOFF RUNWAY LENGTH REQUIREMENTS - STANDARD DAY + 45°F (STD + 25°C) MODEL 747-8F AND 747-8I

PRELIMINARY



3.3.4 F.A.R. TAKEOFF RUNWAY LENGTH REQUIREMENTS – STANDARD DAY + 63 ° F (STD + 35 ° C) MODEL 747-8F AND 747-8I

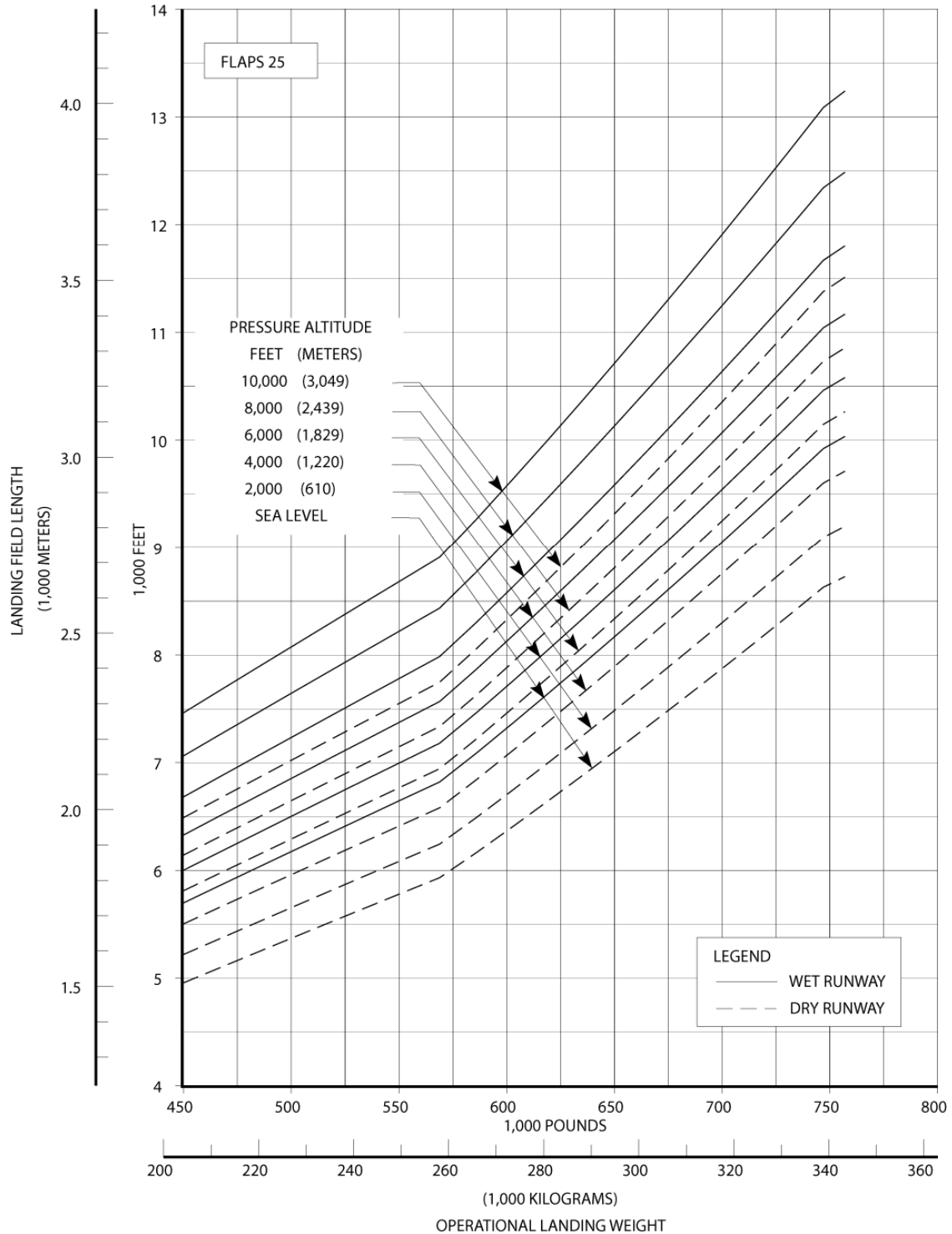
D6-58326-3

PRELIMINARY

STANDARD DAY, ZERO WIND
AUTO SPOILERS OPERATIVE
ANTI-SKID OPERATIVE
ZERO RUNWAY GRADIENT
AIR CONDITIONING OFF
FORWARD CG LIMIT

DO NOT USE FOR DISPATCH

Landing Field Length
747-8F/GENx-2B67

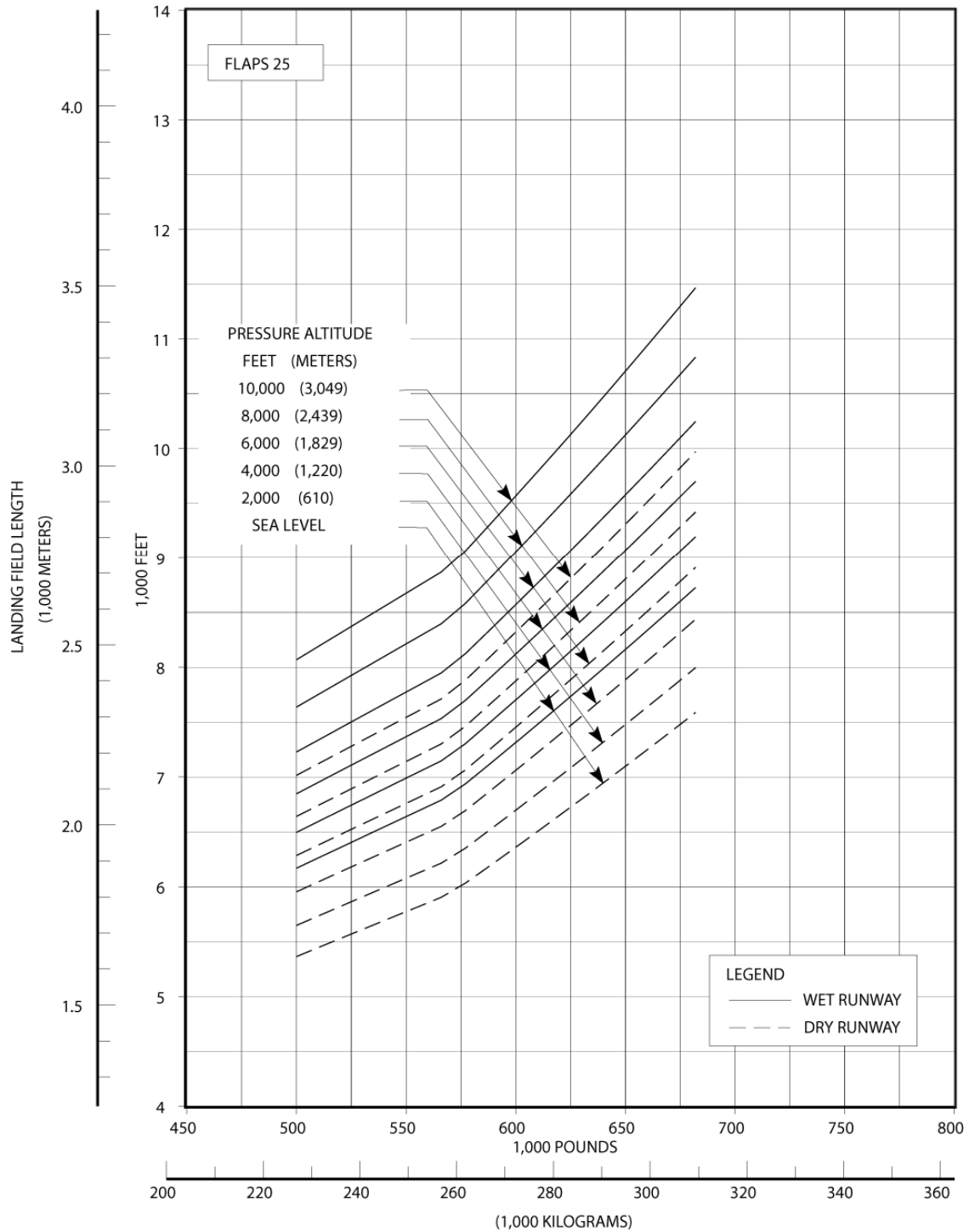


3.4.1 F.A.R. LANDING RUNWAY LENGTH REQUIREMENTS - FLAPS 25 MODEL 747-8F

PRELIMINARY

STANDARD DAY, ZERO WIND
AUTO SPOILERS OPERATIVE
ANTI-SKID OPERATIVE
ZERO RUNWAY GRADIENT
AIR CONDITIONING OFF
FORWARD CG LIMIT

Landing Field Length
747-8I/GENx-2B67



3.4.2 F.A.R. LANDING RUNWAY LENGTH REQUIREMENTS - FLAPS 30 MODEL 747-8I

D6-58326-3

PRELIMINARY

4.0 GROUND MANEUVERING

4.1 General Information

4.2 Turning Radii

4.3 Clearance Radii

4.4 Visibility from Cockpit in Static Position

4.5 Runway and Taxiway Turn Paths

4.6 Runway Holding Bay

PRELIMINARY

4.0 GROUND MANEUVERING

4.1 General Information

4.2 Turning Radii

4.3 Clearance Radii

4.4 Visibility from Cockpit in Static Position

4.5 Runway and Taxiway Turn Paths

4.6 Runway Holding Bay

PRELIMINARY

4.0 GROUND MANEUVERING

4.1 General Information

The 747-8 main landing gear consists of four main struts, each strut with four wheels. This geometric arrangement of the four main gears results in somewhat different ground maneuvering characteristics from those experienced with typical landing gear aircraft.

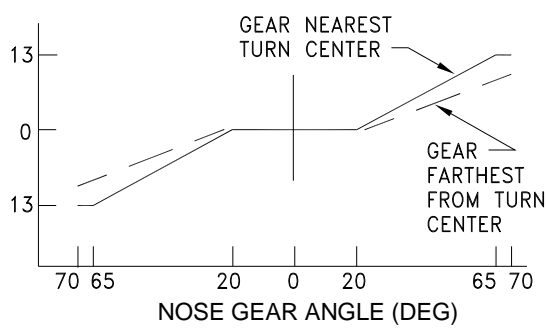
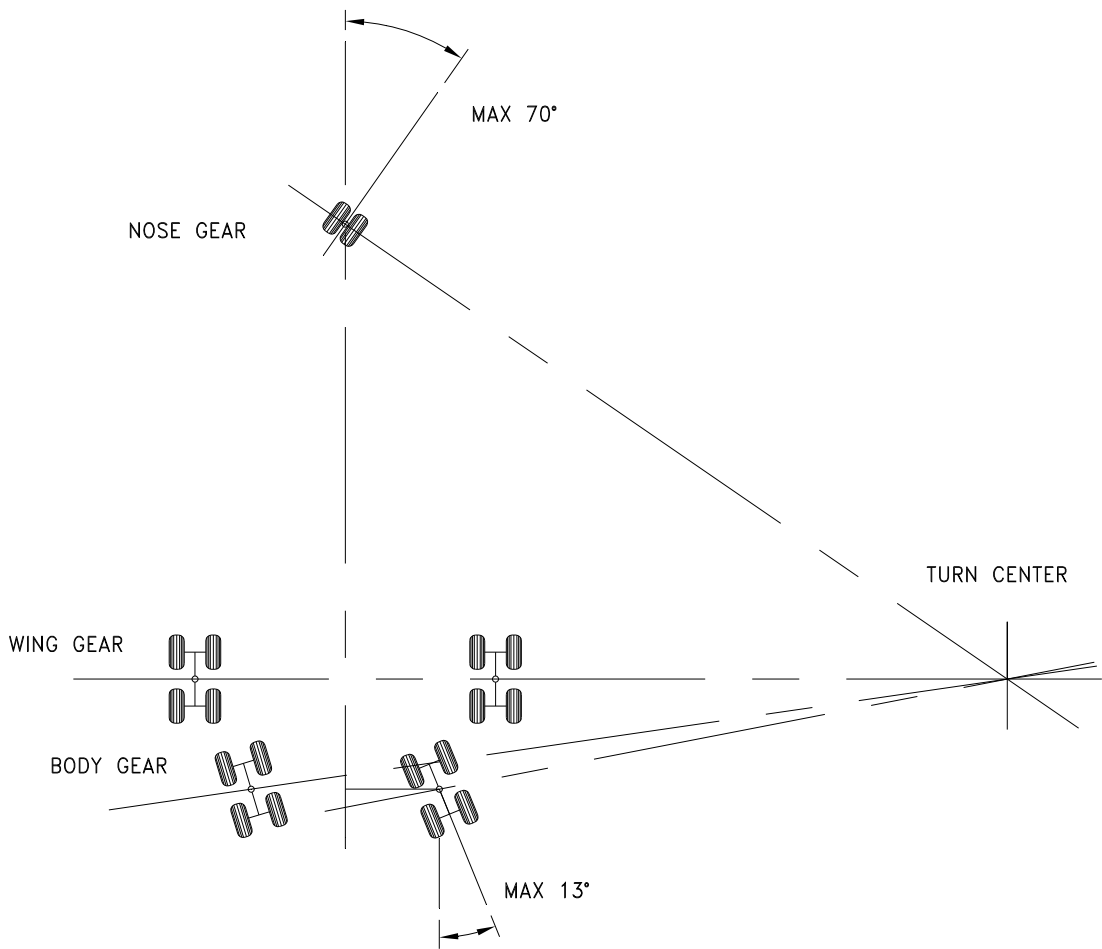
Basic factors that influence the geometry of the turn include:

1. Nose wheel steering angle
2. Engine power settings
3. Center of gravity location
4. Airplane weight
5. Pavement surface conditions
6. Amount of differential braking
7. Ground speed
8. Main landing gear steering

The steering system of the 747-8 incorporates steering of the main body landing gear in addition to the nose gear steering. This body gear steering system is hydraulically actuated and is programmed electrically to provide steering ratios proportionate to the nose gear steering angles. During takeoff and landing, the body gear steering system is centered, mechanically locked, and depressurized.

Steering of the main body gear has the following advantages over ground maneuvering without this steering feature; overall improved maneuverability, including improved nose gear tracking; elimination of the need for differential braking during ground turns, with subsequent reduced brake wear; reduced thrust requirements; lower main gear stress levels; and reduced tire scrubbing. The turning radii shown in Section 4.2 are derived from a previous test involving a 747-200. The 747-8 is expected to follow the same maneuvering characteristics.

PRELIMINARY



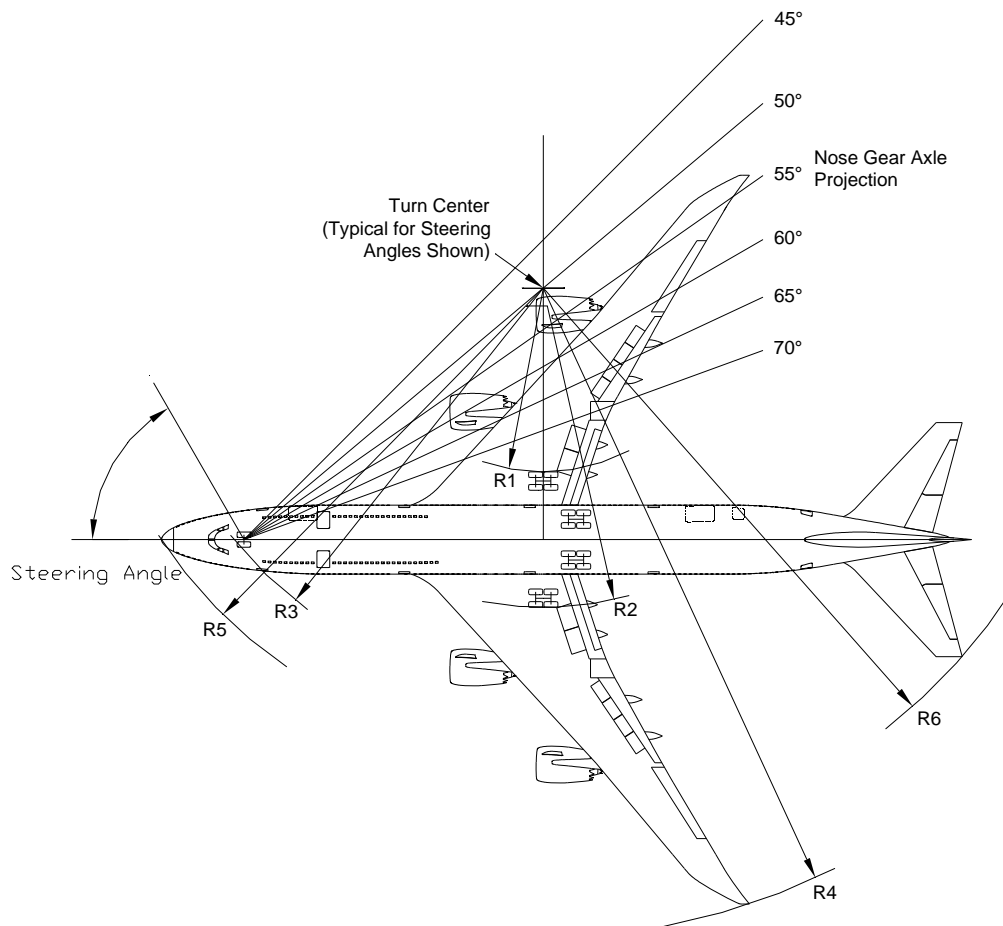
NOSE GEAR	BODY GEAR
0° TO 20°	0°
20° TO 70°	0° TO 13°

NOSE GEAR/BODY GEAR TURN RATIOS

4.1.1 GENERAL INFORMATION – BODY GEAR STEERING SYSTEM

MODEL 747-8, 747-8F

PRELIMINARY



NOTES: DATA SHOWN FOR AIRPLANE WITH BODY GEAR STEERING
 ACTUAL OPERATING TURNING RADII MAY BE GREATER THAN SHOWN
 CONSULT WITH AIRLINE FOR SPECIFIC OPERATING PROCEDURE
 DIMENSIONS ROUNDED TO NEAREST FOOT AND 0.1 METER

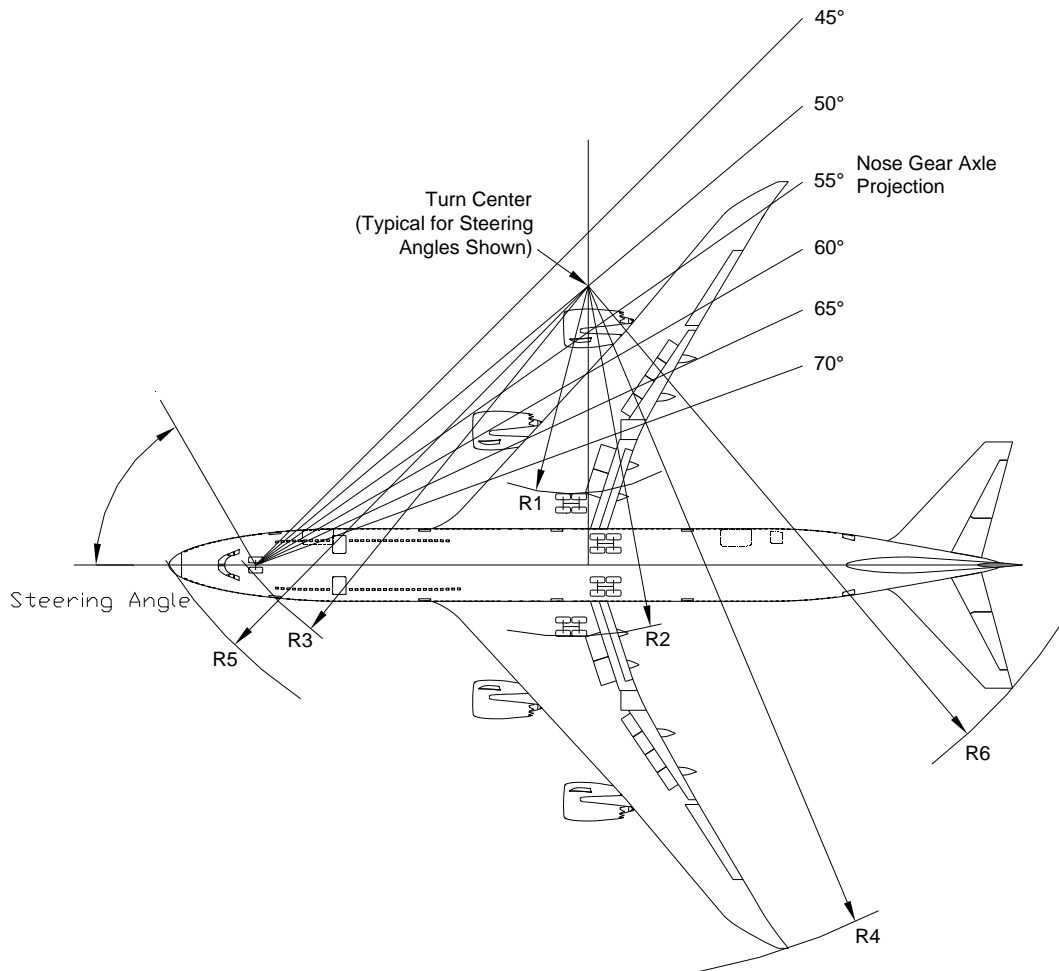
STEERING ANGLE	R1 INNER GEAR		R2 OUTER GEAR		R3 NOSE GEAR		R4 WING TIP		R5 NOSE		R6 TAIL	
(DEG)	FT	M	FT	M	FT	M	FT	M	FT	M	FT	M
30	139	42.4	181	55.2	188	57.3	280	85.3	199	60.7	233	71.0
35	111	33.8	153	46.6	164	50.0	252	76.8	177	53.9	210	64.0
40	89	27.1	131	39.9	147	44.8	231	70.4	161	49.1	193	58.8
45	72	21.9	113	34.4	134	40.8	214	65.2	150	45.7	180	54.9
50	57	17.4	98	29.9	124	37.8	200	61.0	141	43.0	170	51.8
55	44	13.4	86	26.2	116	35.4	188	57.3	134	40.8	162	49.4
60	33	10.1	74	22.6	110	33.5	177	53.9	129	39.3	155	47.2
65	22	6.7	64	19.5	105	32.0	168	51.2	125	38.1	149	45.4
70 (MAX)	13	4.0	55	16.8	101	30.8	159	48.5	123	37.5	144	43.9

4.2.1 TURNING RADII – NO SLIP ANGLE – WITH BODY GEAR STEERING

D6-58326-3

PRELIMINARY

MODEL 747-8, 747-8F

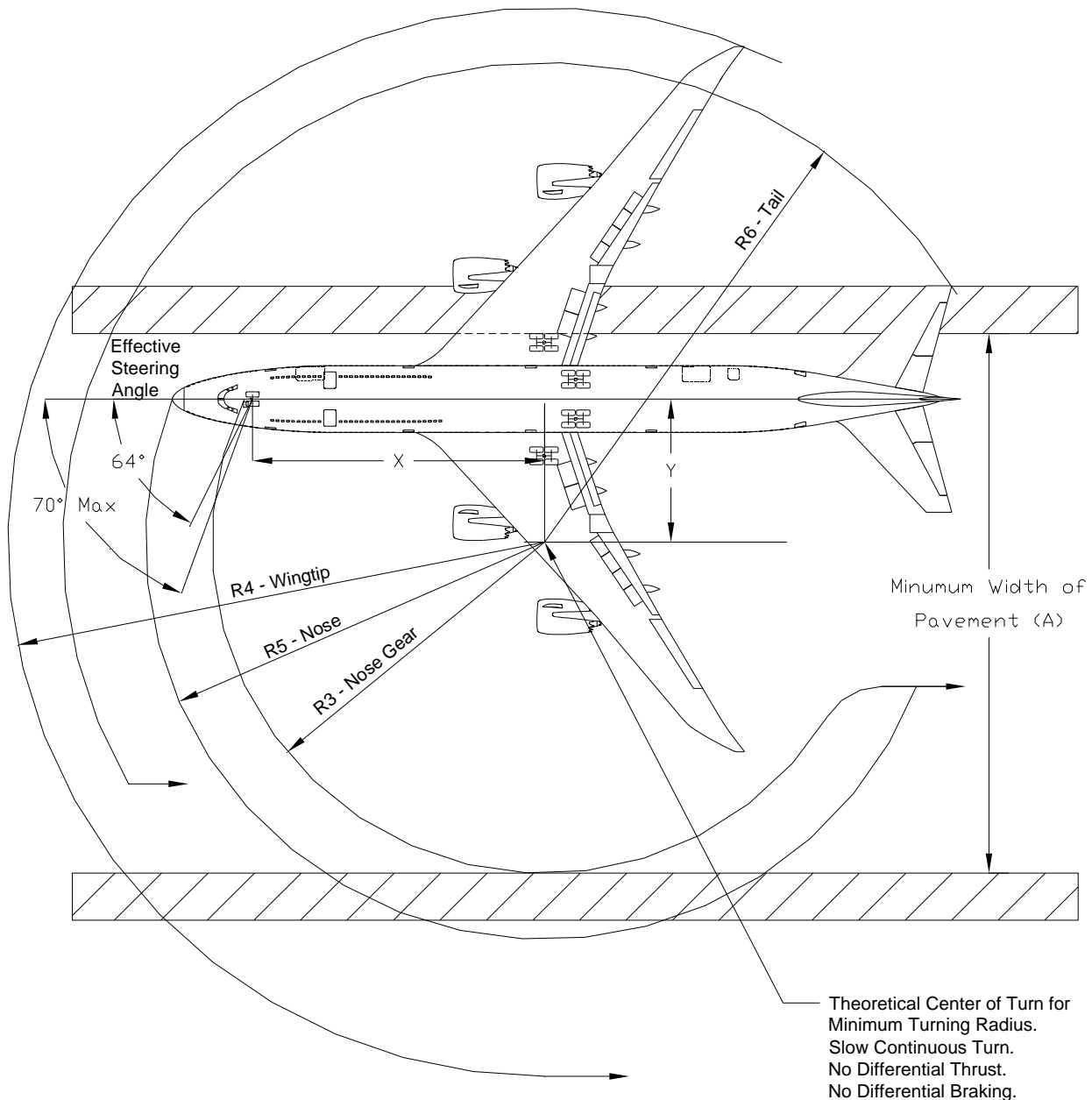


NOTES: DATA SHOWN FOR AIRPLANE WITH BODY GEAR STEERING INOPERATIVE
 ACTUAL OPERATING TURNING RADII MAY BE GREATER THAN SHOWN
 CONSULT WITH AIRLINE FOR SPECIFIC OPERATING PROCEDURE
 DIMENSIONS ROUNDED TO NEAREST FOOT AND 0.1 METER

STEERING ANGLE	R1		R2		R3		R4		R5		R6	
(DEG)	INNER	GEAR	OUTER	GEAR	NOSE	GEAR	WING	TIP	NOSE		TAIL	
	FT	M	FT	M	FT	M	FT	M	FT	M	FT	M
30	148	45.1	190	57.9	198	60.4	287	87.5	209	63.7	240	73.2
35	118	36.0	160	48.8	173	52.7	258	78.6	186	56.7	215	65.5
40	95	29.0	137	41.8	155	47.2	236	71.9	169	51.5	196	59.7
45	77	23.5	118	36.0	141	43.0	218	66.4	157	47.9	182	55.5
50	61	18.6	103	31.4	130	39.6	203	61.9	148	45.1	171	52.1
55	47	14.3	89	27.1	122	37.2	190	57.9	141	43.0	162	49.4
60	36	11.0	77	23.5	116	35.4	178	54.3	135	41.1	155	47.2
65	25	7.6	66	20.1	111	33.8	168	51.2	131	39.9	149	45.4
70 (MAX)	15	4.6	57	17.4	107	32.6	159	48.5	128	39.0	143	43.6

4.2.2 TURNING RADII – NO SLIP ANGLE –BODY GEAR STEERING INOPERATIVE MODEL 747-8, 747-8F

PRELIMINARY



Notes:

- 6° Tire Slip Angle – Approximate Only For 70° Maximum Turn Angle
- Consult Airline For Actual Operating Data.

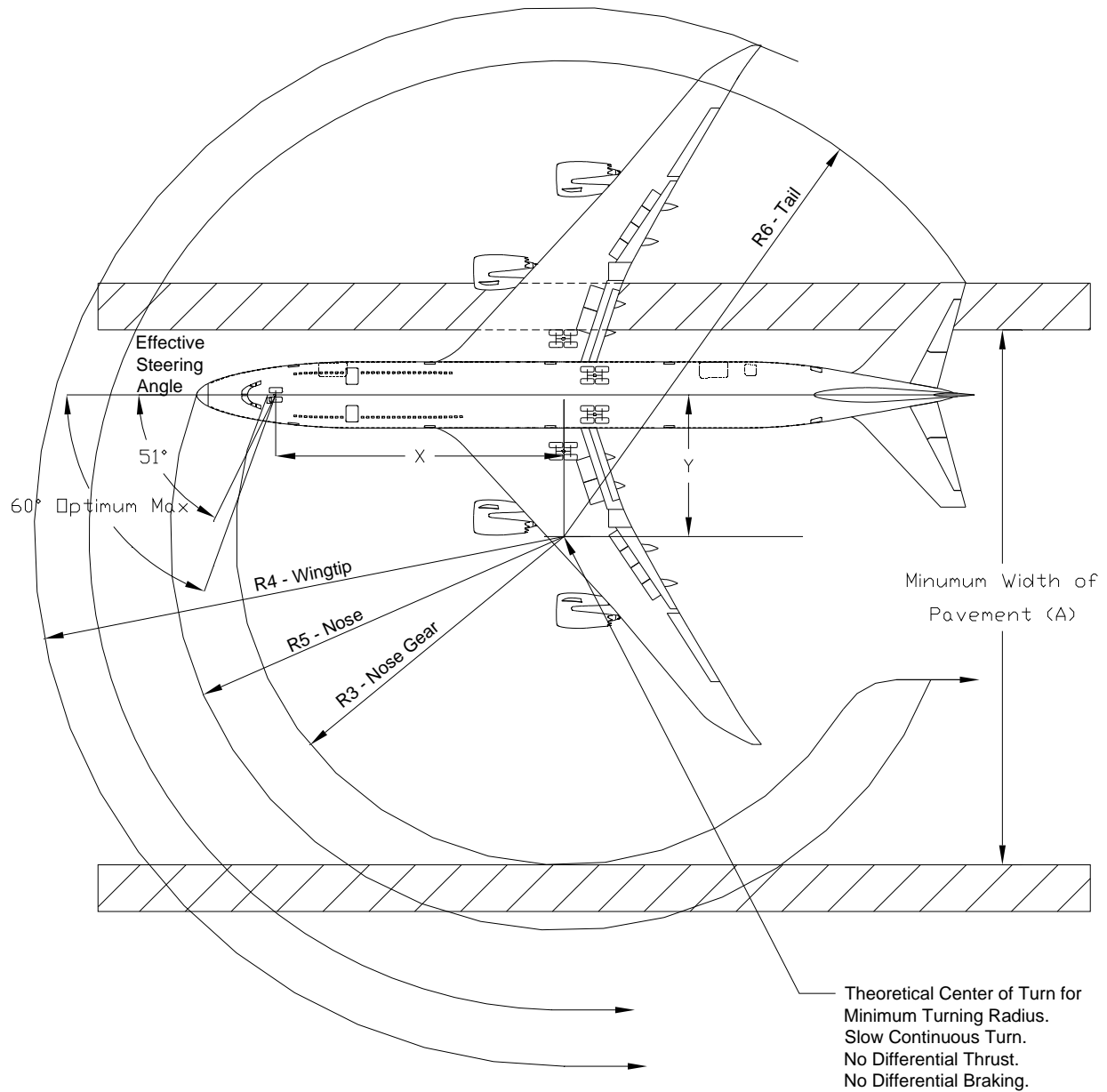
AIRPLANE MODEL	EFFECTIVE TURNING ANGLE (DEG)	X		Y		A		R3		R4		R5		R6	
		FT	M	FT	M	FT	M	FT	M	FT	M	FT	M	FT	M
747-8F	64	92	28.0	45	13.7	172	52.4	106	32.3	170	51.8	126	38.4	150	45.7

NOTE: DIMENSIONS ARE ROUNDED TO THE NEAREST FOOT AND 0.1 METER.

4.3.1 CLEARANCE RADII – WITH BODY GEAR STEERING MODEL 747-8, 747-8F

D6-58326-3

PRELIMINARY



Notes:

- **Body Gear Steering Inoperative Rarely Occurs. Data Provided As Reference Only**
- **9° Tire Slip Angle – Approximate Only For 60° Turn Angle (Optimum Max Steering Angle)**
- **Consult Airline For Actual Operating Data.**

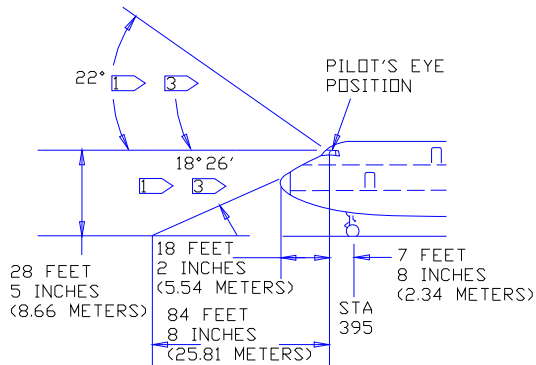
AIRPLANE MODEL	EFFECTIVE TURNING ANGLE (DEG)	X		Y		A		R3		R4		R5		R6	
		FT	M	FT	M	FT	M	FT	M	FT	M	FT	M	FT	M
747-8F	51	98	29.9	79	24.1	228	69.5	129	39.3	200	61.0	146	44.5	169	51.5

NOTE: DIMENSIONS ARE ROUNDED TO THE NEAREST FOOT AND 0.1 METER.

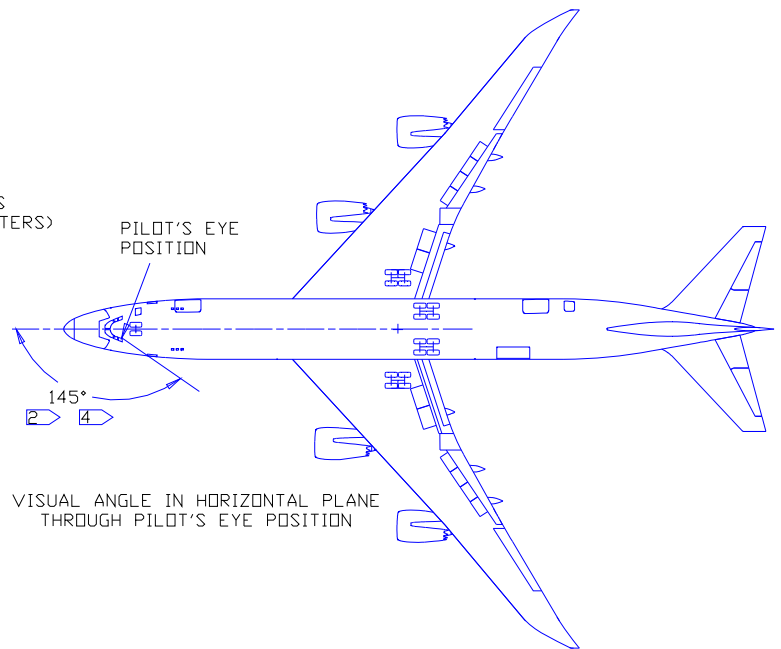
4.3.2 CLEARANCE RADII –BODY GEAR STEERING INOPERATIVE MODEL 747-8, 747-8F

PRELIMINARY

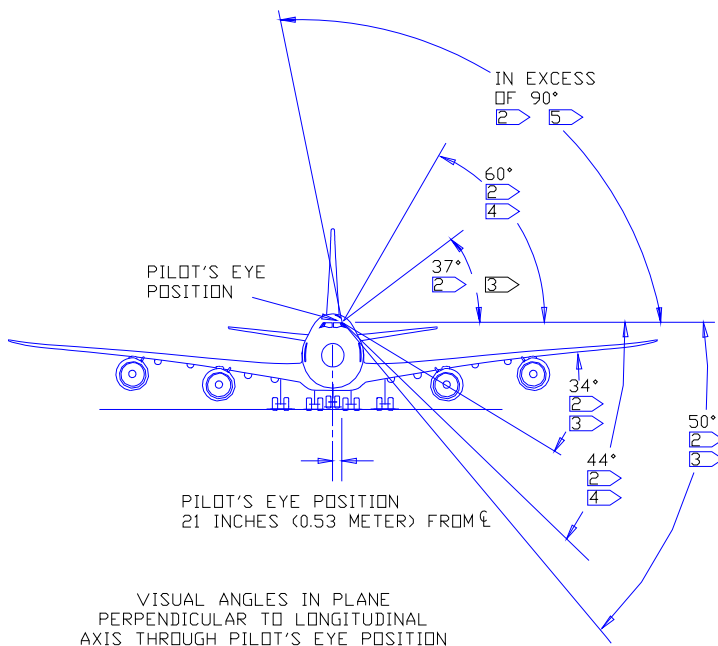
NOT TO BE USED FOR
LANDING APPROACH
VISIBILITY



VISUAL ANGLES IN VERTICAL PLANE
THROUGH PILOT'S EYE POSITION



VISUAL ANGLE IN HORIZONTAL PLANE
THROUGH PILOT'S EYE POSITION



VISUAL ANGLES IN PLANE
PERPENDICULAR TO LONGITUDINAL
AXIS THROUGH PILOT'S EYE POSITION

NOTES:

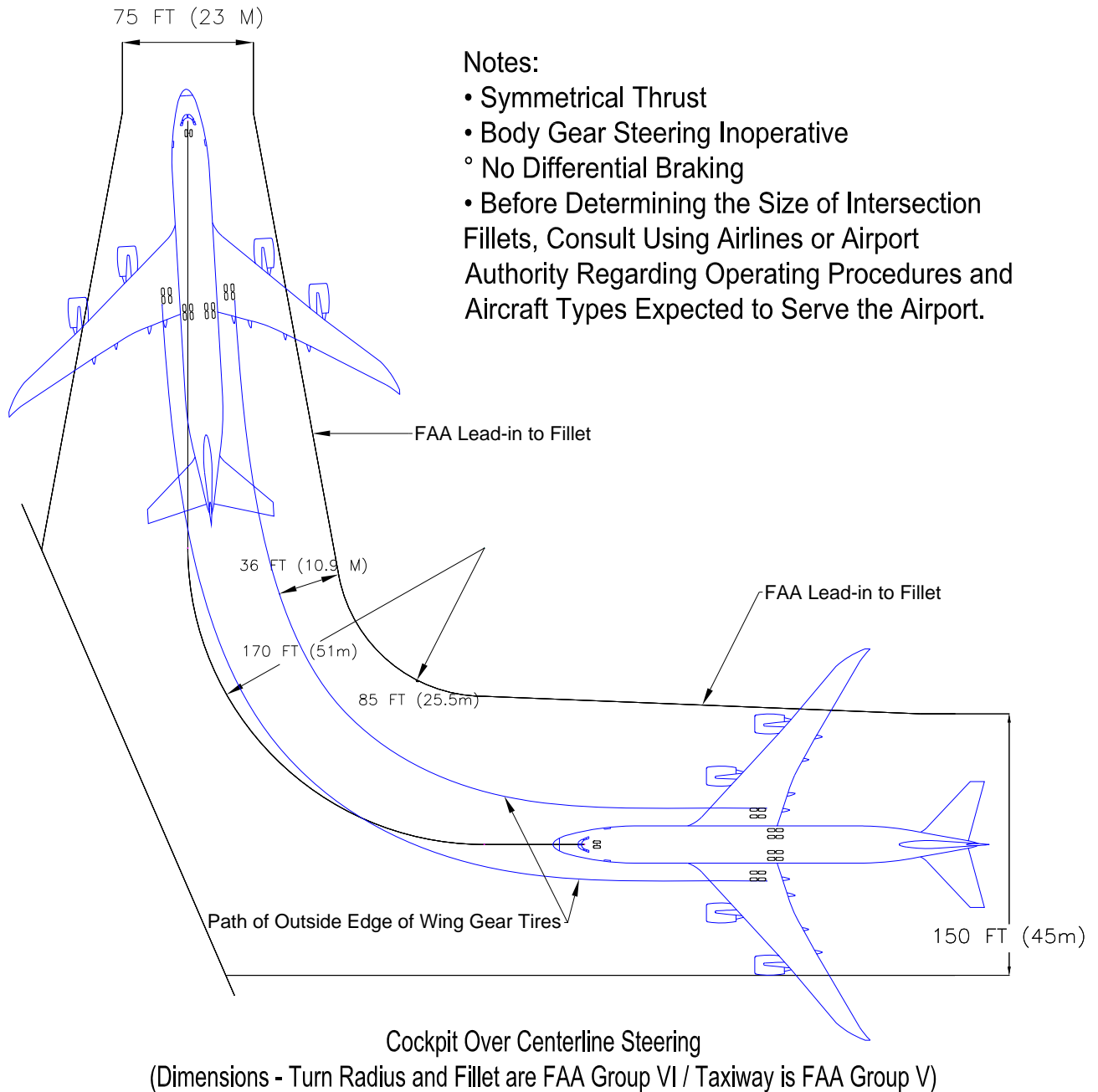
- 1 VISUAL ANGLES THROUGH WINDSHIELD
- 2 VISUAL ANGLES THROUGH SIDE WINDOW
- 3 VISUAL ANGLES FROM NORMAL POSITION
- 4 VISUAL ANGLES FROM ALERT POSITION. HEAD MOVED OUTBOARD 5 INCHES (0.13 METERS)
- 5 VISUAL ANGLES WITH HEAD MOVED OUTBOARD 7 INCHES (0.18 METERS)

* HEAD IS ROTATED ABOUT A
POINT 3 IN (0.08 METER) AFT
OF PILOT'S EYE POSITION

4.4 VISIBILITY FROM COCKPIT IN STATIC POSITION MODEL 747-8, 747-8F

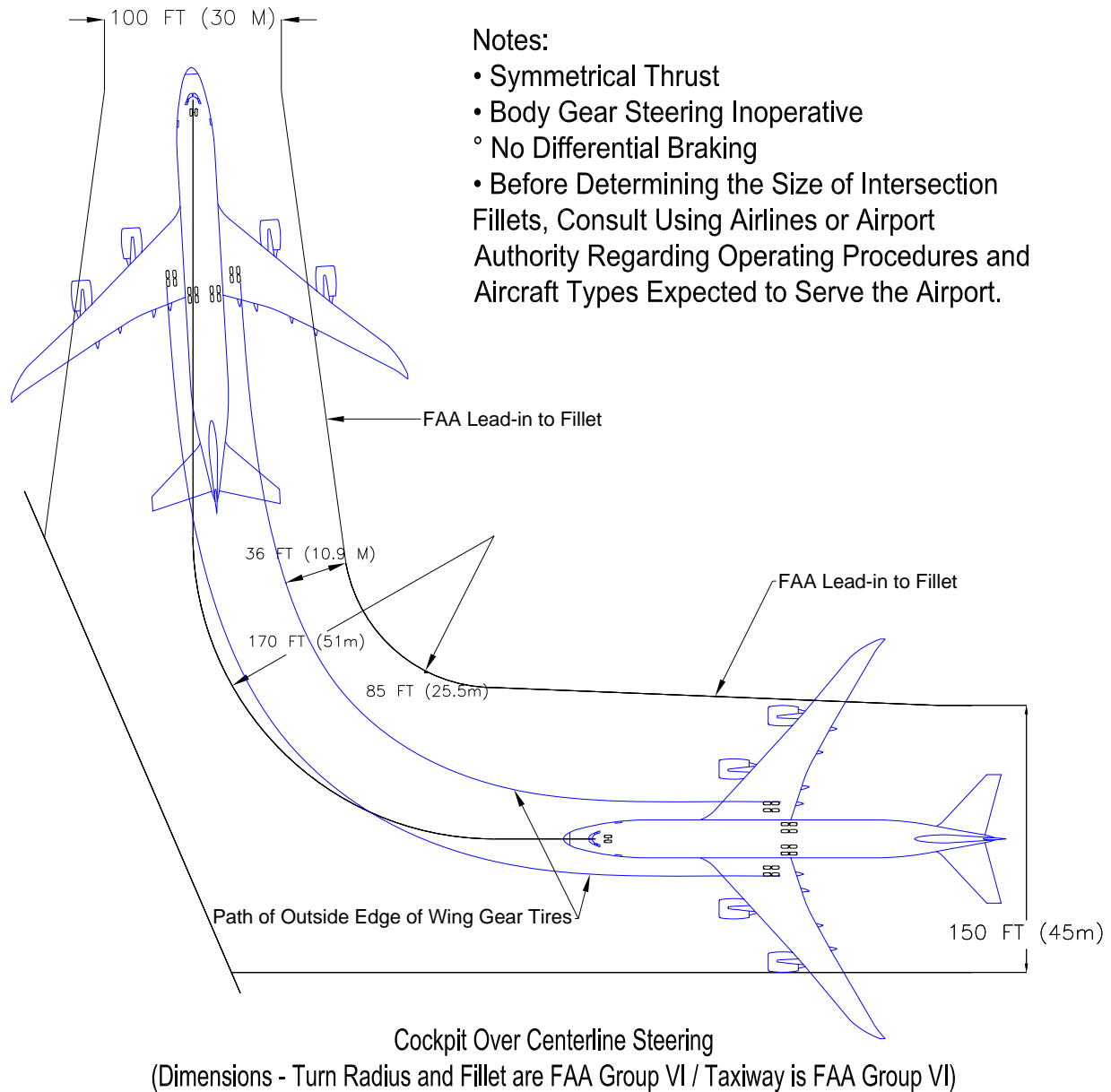
D6-58326-3

PRELIMINARY



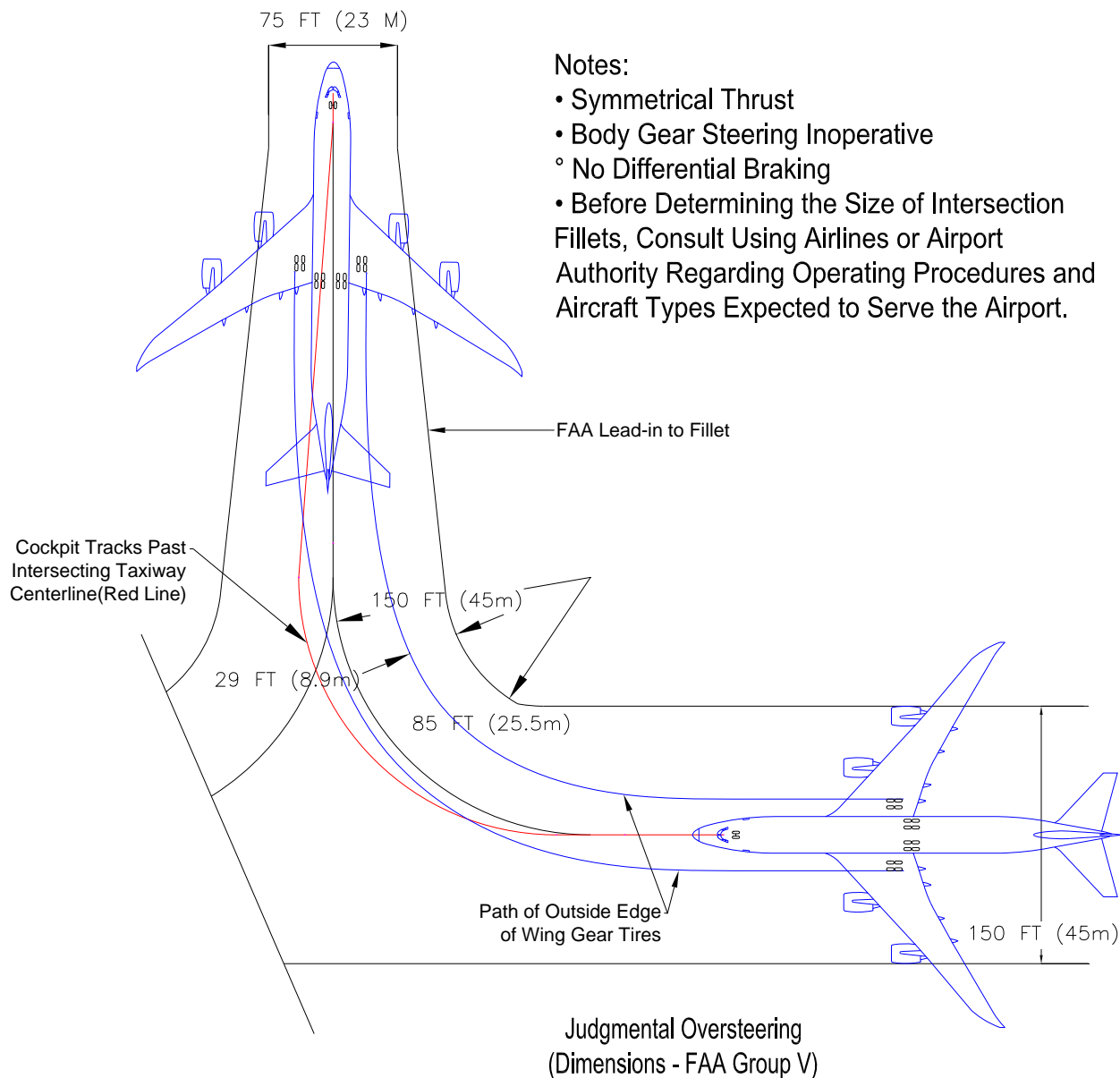
4.5.1 RUNWAY AND TAXIWAY TURNPATHS - RUNWAY-TO-TAXIWAY, 90 DEGREES, COCKPIT OVER CENTERLINE (FAA GROUP VI RADIUS/FILLET TO GROUP V TAXIWAY) MODEL 747-8, 747-8F

PRELIMINARY



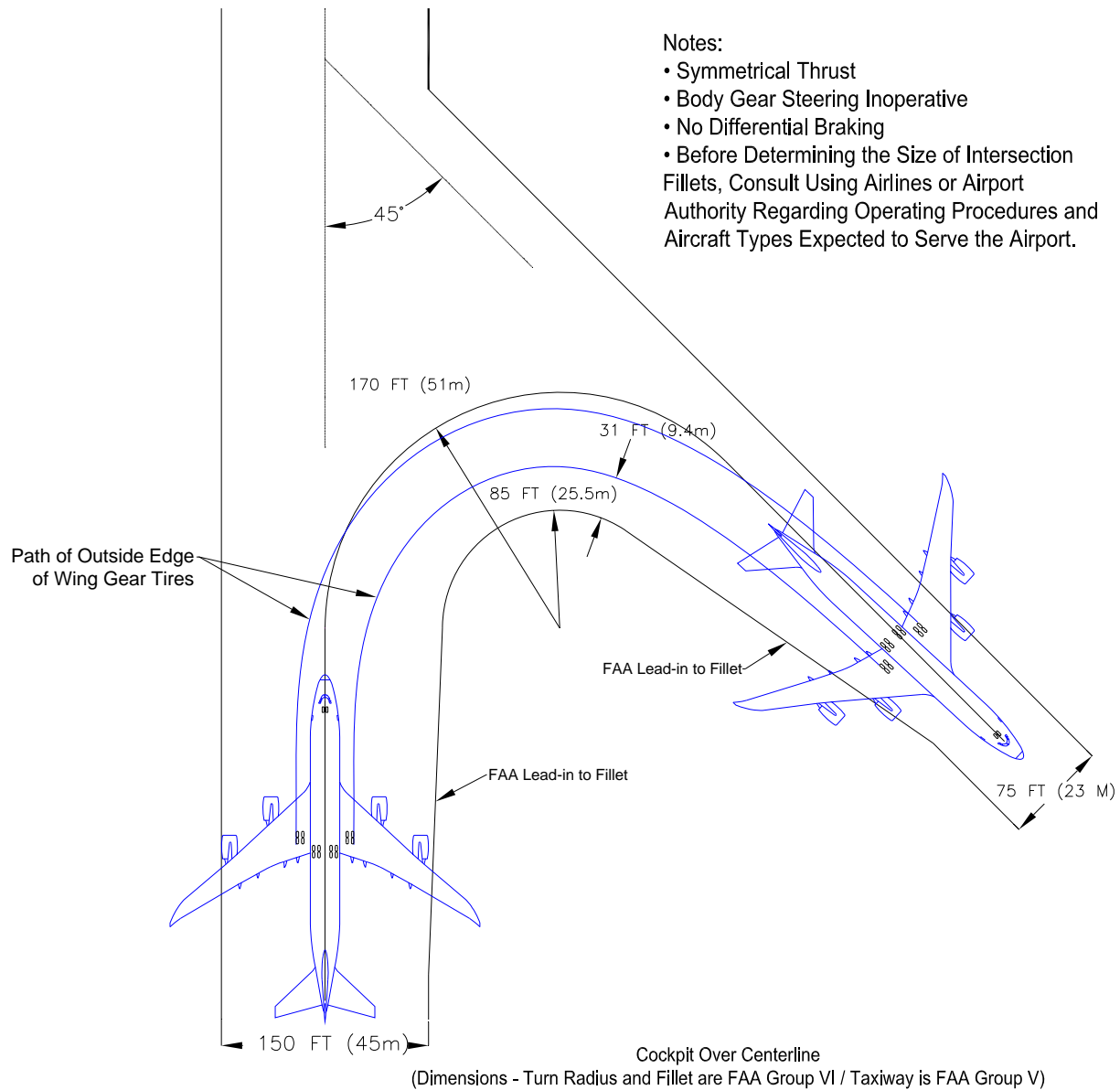
4.5.2 RUNWAY AND TAXIWAY TURNPATHS - RUNWAY-TO-TAXIWAY, 90 DEGREES, COCKPIT OVER CENTERLINE (FAA GROUP VI RADIUS/FILLET TO GROUP VI TAXIWAY) MODEL 747-8, 747-8F

PRELIMINARY



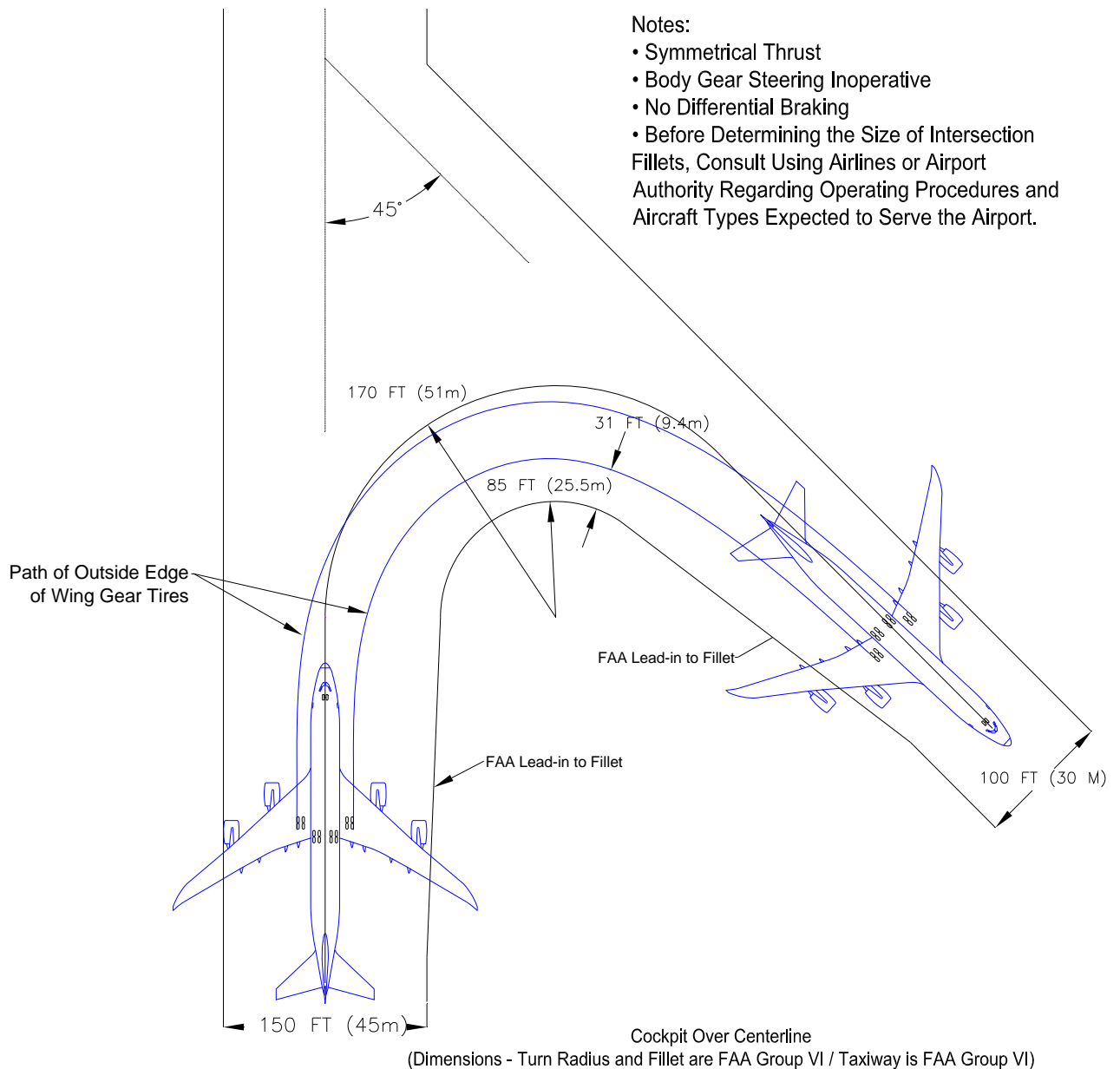
4.5.3 RUNWAY AND TAXIWAY TURNPATHS - RUNWAY-TO-TAXIWAY, 90 DEGREES, JUDGMENTAL OVERSTEER (FAA GROUP V RADIUS/FILLET TO GROUP V TAXIWAY) MODEL 747-8, 747-8F

PRELIMINARY



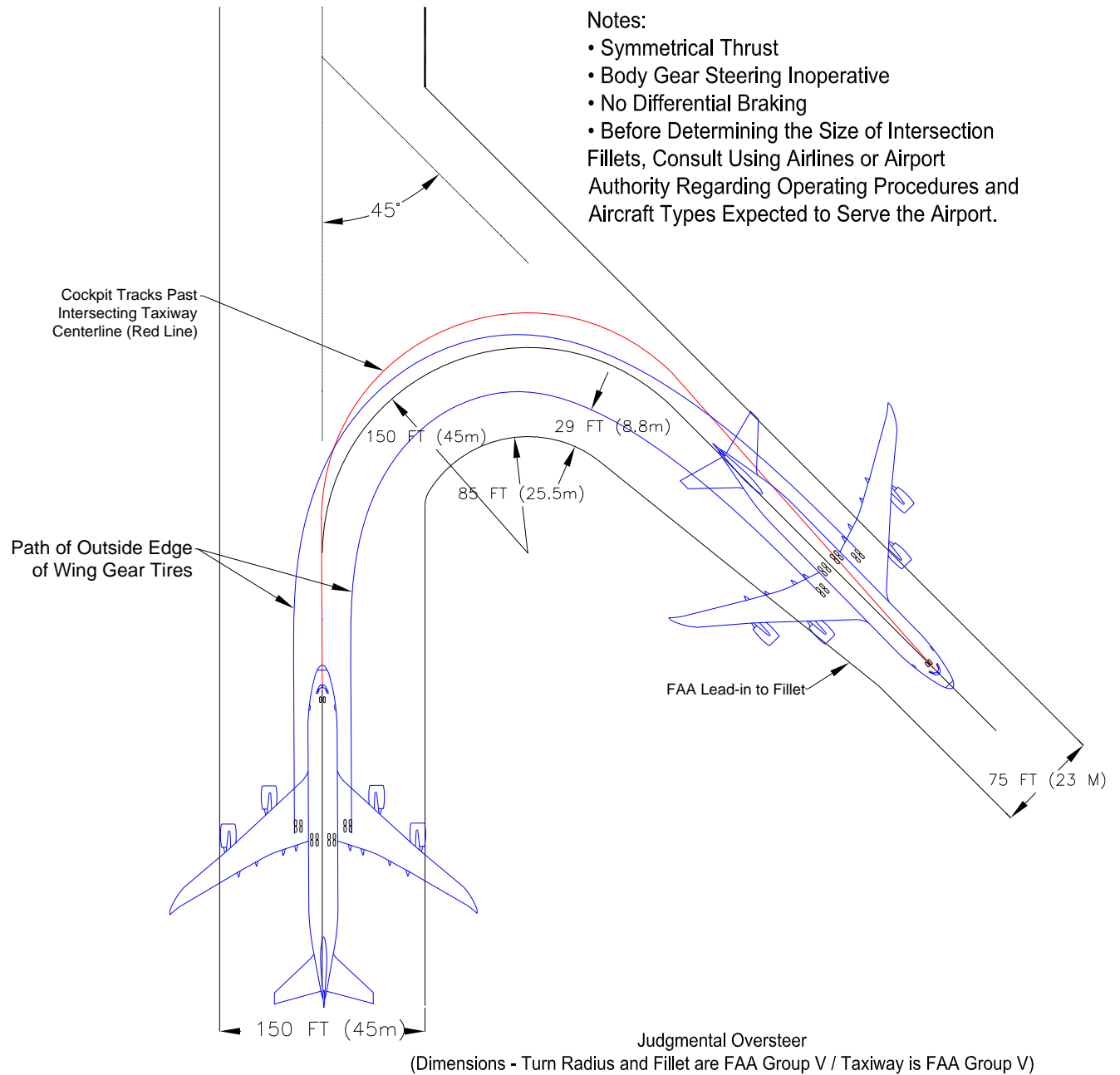
4.5.4 RUNWAY AND TAXIWAY TURNPATHS - RUNWAY-TO-TAXIWAY, MORE THAN 90 DEGREES, COCKPIT OVER CENTERLINE (FAA GROUP VI RADIUS TO GROUP V TAXIWAY) MODEL 747-8, 747-8F

PRELIMINARY



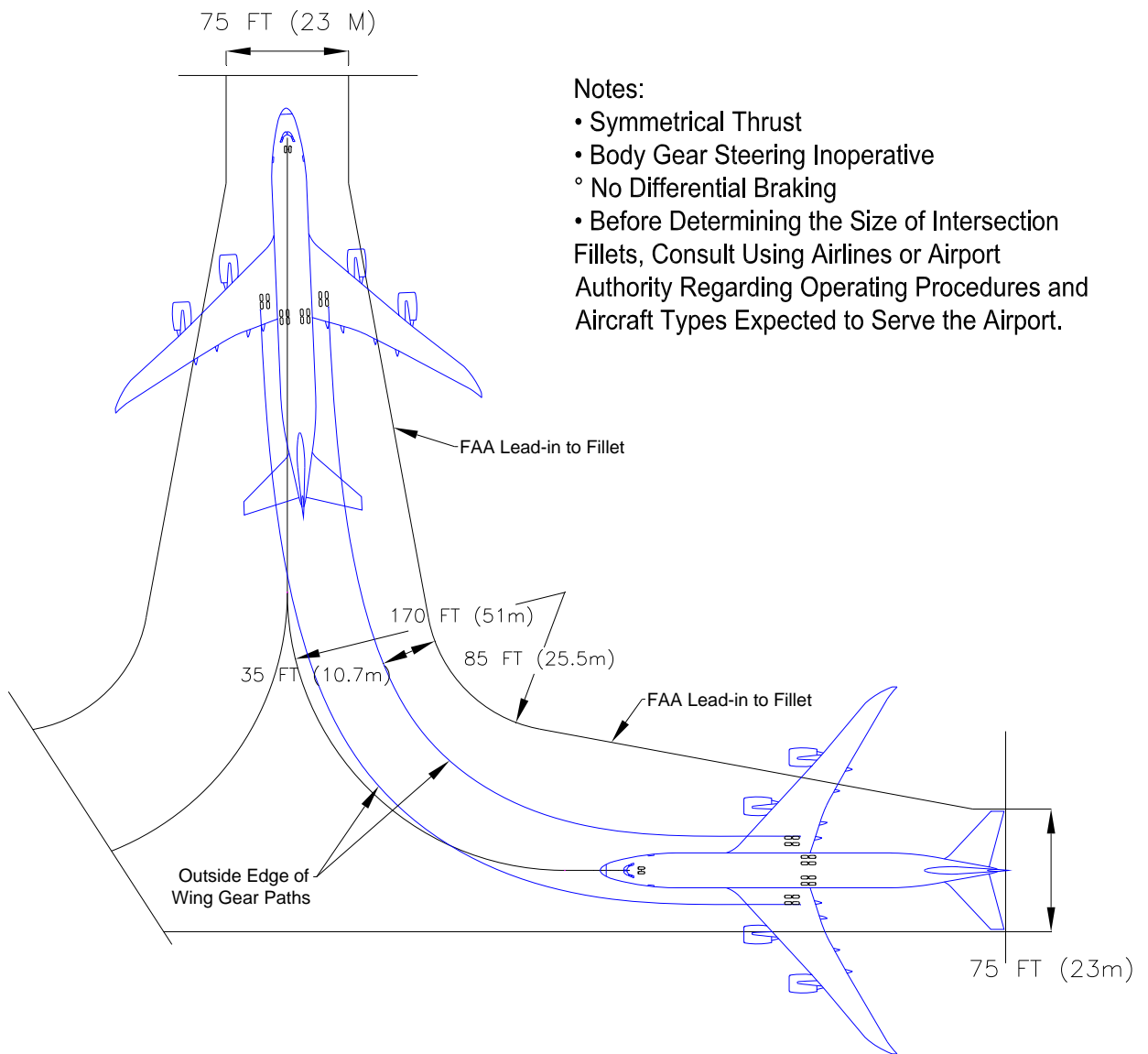
4.5.5 RUNWAY AND TAXIWAY TURNPATHS - RUNWAY-TO-TAXIWAY, MORE THAN 90 DEGREES, COCKPIT OVER CENTERLINE (FAA GROUP VI RADIUS TO GROUP VI TAXIWAY) MODEL 747-8, 747-8F

PRELIMINARY



4.5.6 RUNWAY AND TAXIWAY TURNPATHS - RUNWAY-TO-TAXIWAY, MORE THAN 90 DEGREES, JUDGMENTAL OVERSTEER (FAA GROUP V RADIUS TO GROUP V TAXIWAY) MODEL 747-8, 747-8F

PRELIMINARY



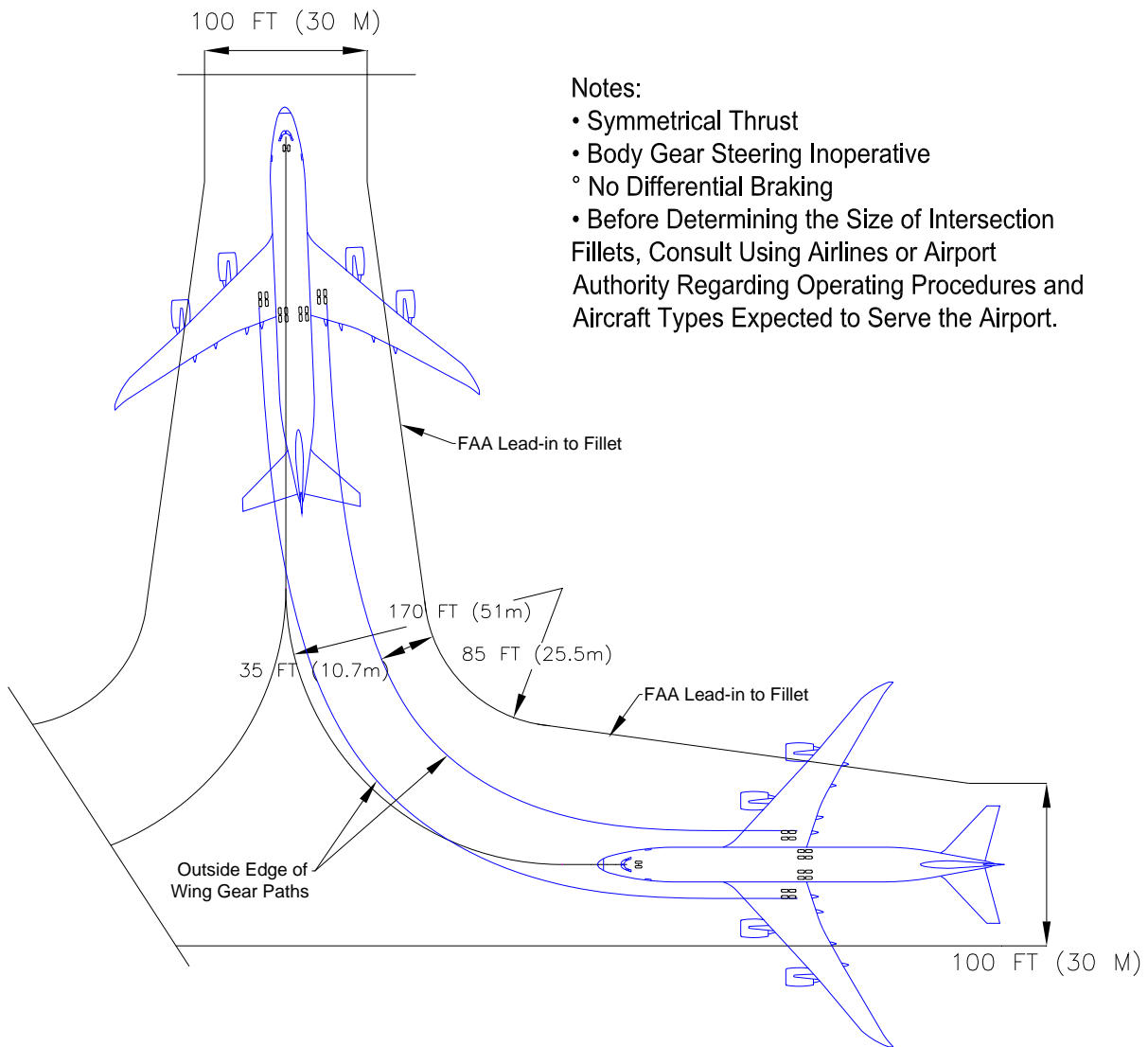
Notes:

- Symmetrical Thrust
- Body Gear Steering Inoperative
- ° No Differential Braking
- Before Determining the Size of Intersection Fillets, Consult Using Airlines or Airport Authority Regarding Operating Procedures and Aircraft Types Expected to Serve the Airport.

Cockpit Over Centerline Steering
(Dimensions - Turn Radius and Fillet are FAA Group VI / Taxiways are FAA Group V)

4.5.7 RUNWAY AND TAXIWAY TURNPATHS - TAXIWAY -TO-TAXIWAY, 90 DEGREES, COCKPIT OVER CENTERLINE (FAA GROUP VI RADIUS TO GROUP V TAXIWAYS) MODEL 747-8, 747-8F

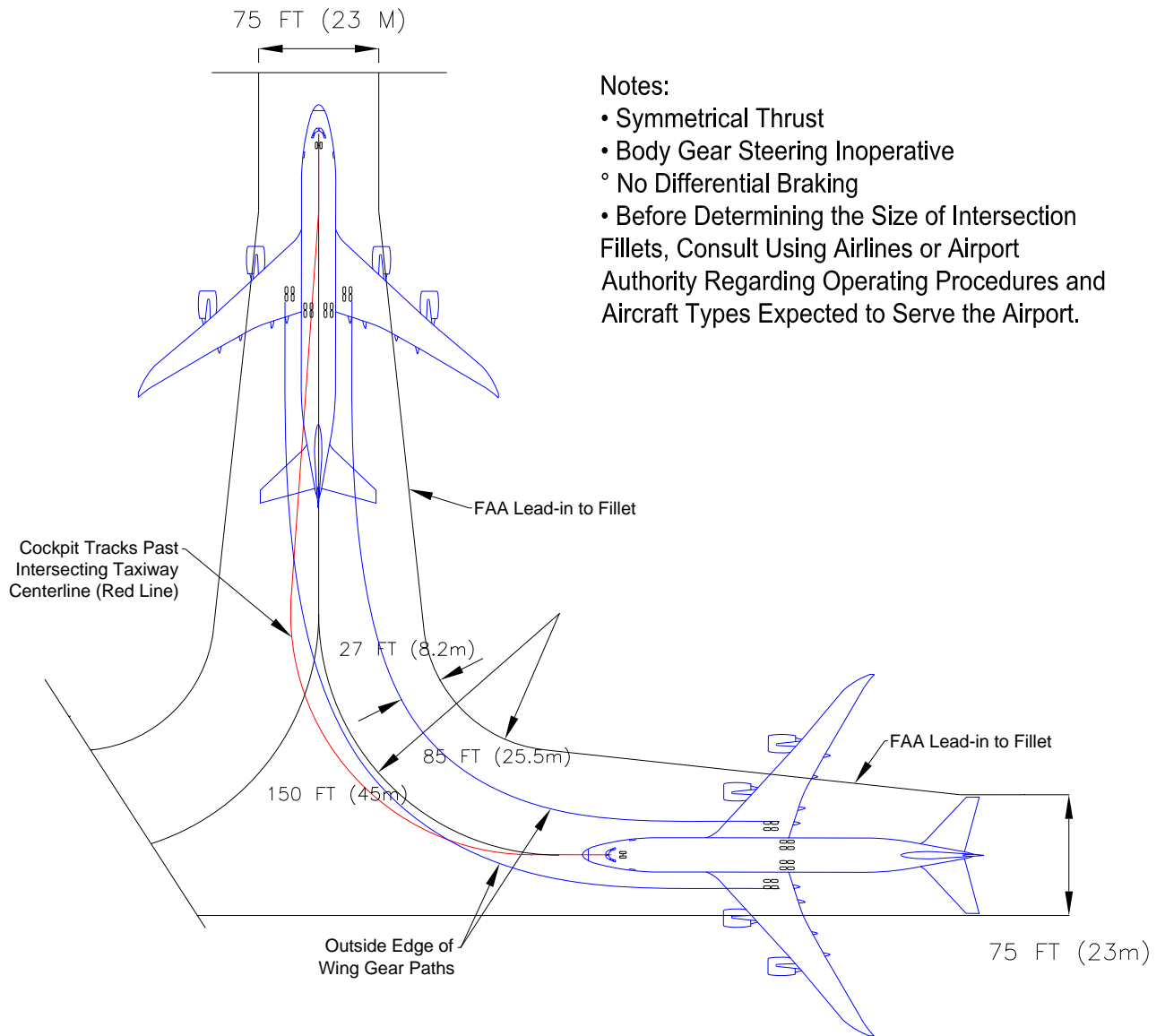
PRELIMINARY



Cockpit Over Centerline Steering
(Dimensions - Turn Radius and Fillet are FAA Group VI / Taxiways are FAA Group VI)

4.5.8 RUNWAY AND TAXIWAY TURNPATHS - TAXIWAY -TO-TAXIWAY, 90 DEGREES, COCKPIT OVER CENTERLINE (FAA GROUP VI RADIUS TO GROUP VI TAXIWAYS) MODEL 747-8, 747-8F

PRELIMINARY



Notes:

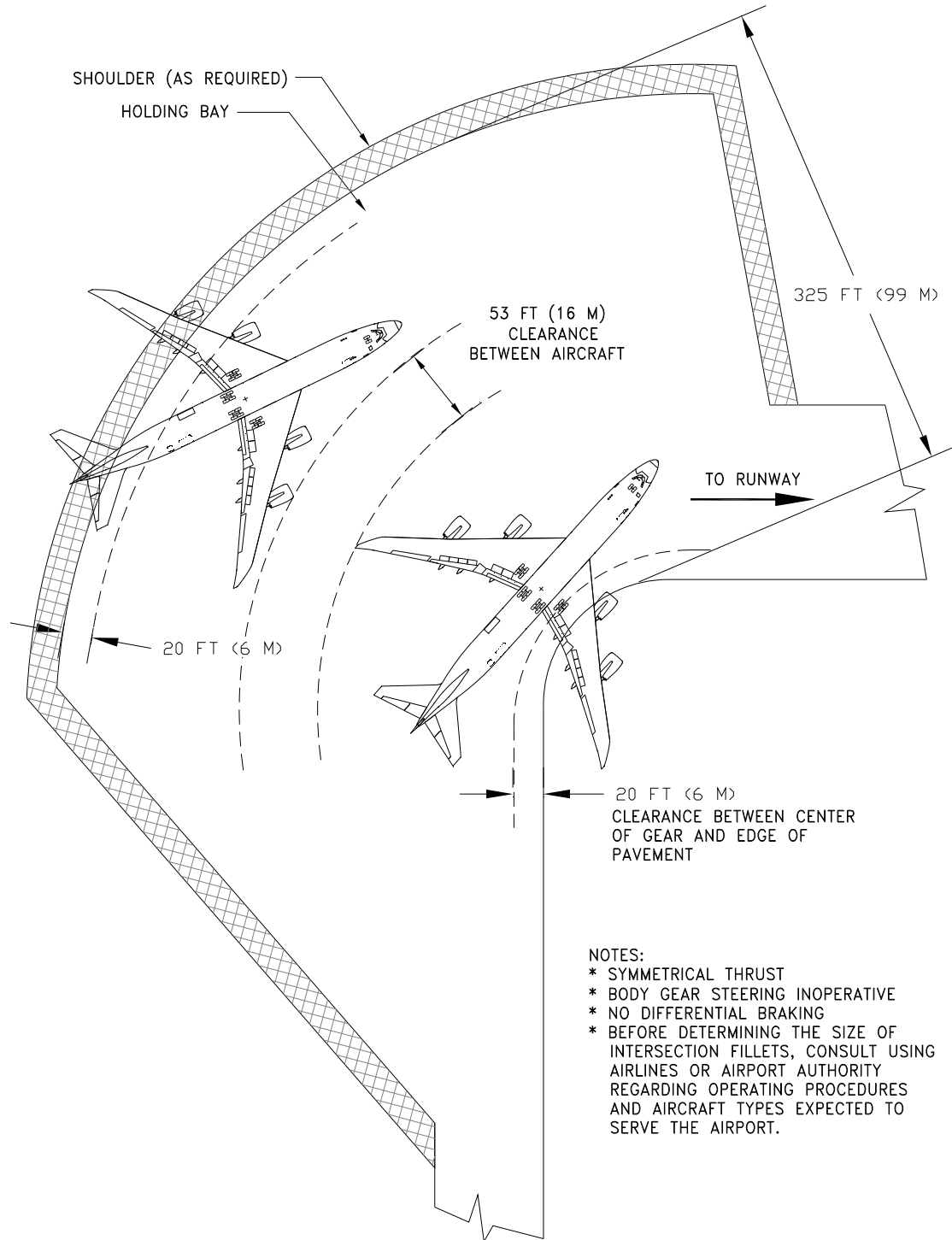
- Symmetrical Thrust
- Body Gear Steering Inoperative
- ° No Differential Braking
- Before Determining the Size of Intersection Fillets, Consult Using Airlines or Airport Authority Regarding Operating Procedures and Aircraft Types Expected to Serve the Airport.

Judgmental Oversteering

(Dimensions - Turn Radius and Fillet are FAA Group V / Taxiways are FAA Group V)

4.5.9 RUNWAY AND TAXIWAY TURNPATHS - TAXIWAY -TO-TAXIWAY, 90 DEGREES, JUDGMENTAL OVERSTEER (FAA GROUP V RADIUS TO GROUP V TAXIWAY) MODEL 747-8, 747-8F

PRELIMINARY



4.6 RUNWAY HOLDING BAY MODEL 747-8, 747-8F

PRELIMINARY

5.0 TERMINAL SERVICING

- 5.1 Airplane Servicing Arrangement - Typical Turnaround**
- 5.2 Terminal Operations - Turnaround Station**
- 5.3 Terminal Operations - En Route Station**
- 5.4 Ground Servicing Connections**
- 5.5 Engine Starting Pneumatic Requirements**
- 5.6 Ground Pneumatic Power Requirements**
- 5.7 Conditioned Air Requirements**
- 5.8 Ground Towing Requirements**

PRELIMINARY

5.0 TERMINAL SERVICING

During turnaround at the terminal, certain services must be performed on the aircraft, usually within a given time, to meet flight schedules. This section shows service vehicle arrangements, schedules, locations of service points, and typical service requirements. The data presented in this section reflect ideal conditions for a single airplane. Service requirements may vary according to airplane condition and airline procedure.

Section 5.1 shows typical arrangements of ground support equipment during turnaround. As noted, if the auxiliary power unit (APU) is used, the electrical, air start, and air-conditioning service vehicles would not be required. Passenger loading bridges or portable passenger stairs could be used to load or unload passengers.

Sections 5.2 and 5.3 show typical service times at the terminal. These charts give typical schedules for performing service on the airplane within a given time. Service times could be rearranged to suit availability of personnel, airplane configuration, and degree of service required.

Section 5.4 shows the locations of ground service connections in graphic and in tabular forms. Typical capacities and service requirements are shown in the tables. Services with requirements that vary with conditions are described in subsequent sections.

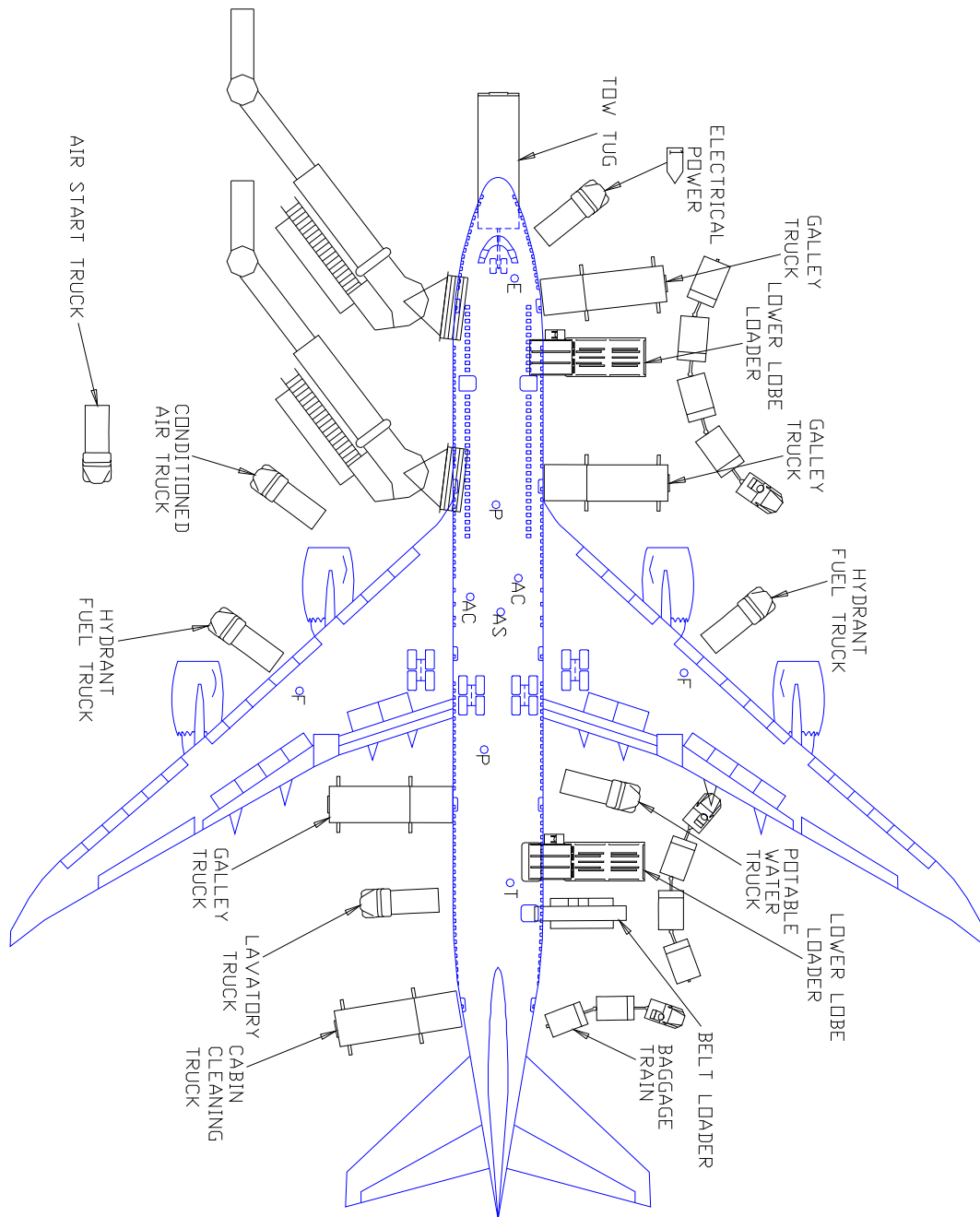
Section 5.5 shows typical sea level air pressure and flow requirements for starting different engines. The curves are based on an engine start time of 90 seconds.

Section 5.6 shows pneumatic requirements for heating and cooling (air conditioning) using high pressure air to run the air cycle machine. The curves show airflow requirements to heat or cool the airplane within a given time and ambient conditions. Maximum allowable pressure and temperature for air cycle machine operation are 60 psia and 450°F, respectively.

Section 5.7 shows pneumatic requirements for heating and cooling the airplane, using low pressure conditioned air. This conditioned air is supplied through an 8-in ground air connection (GAC) directly to the passenger cabin, bypassing the air cycle machines.

Section 5.8 shows ground towing requirements for various ground surface conditions.

PRELIMINARY



LEGEND:

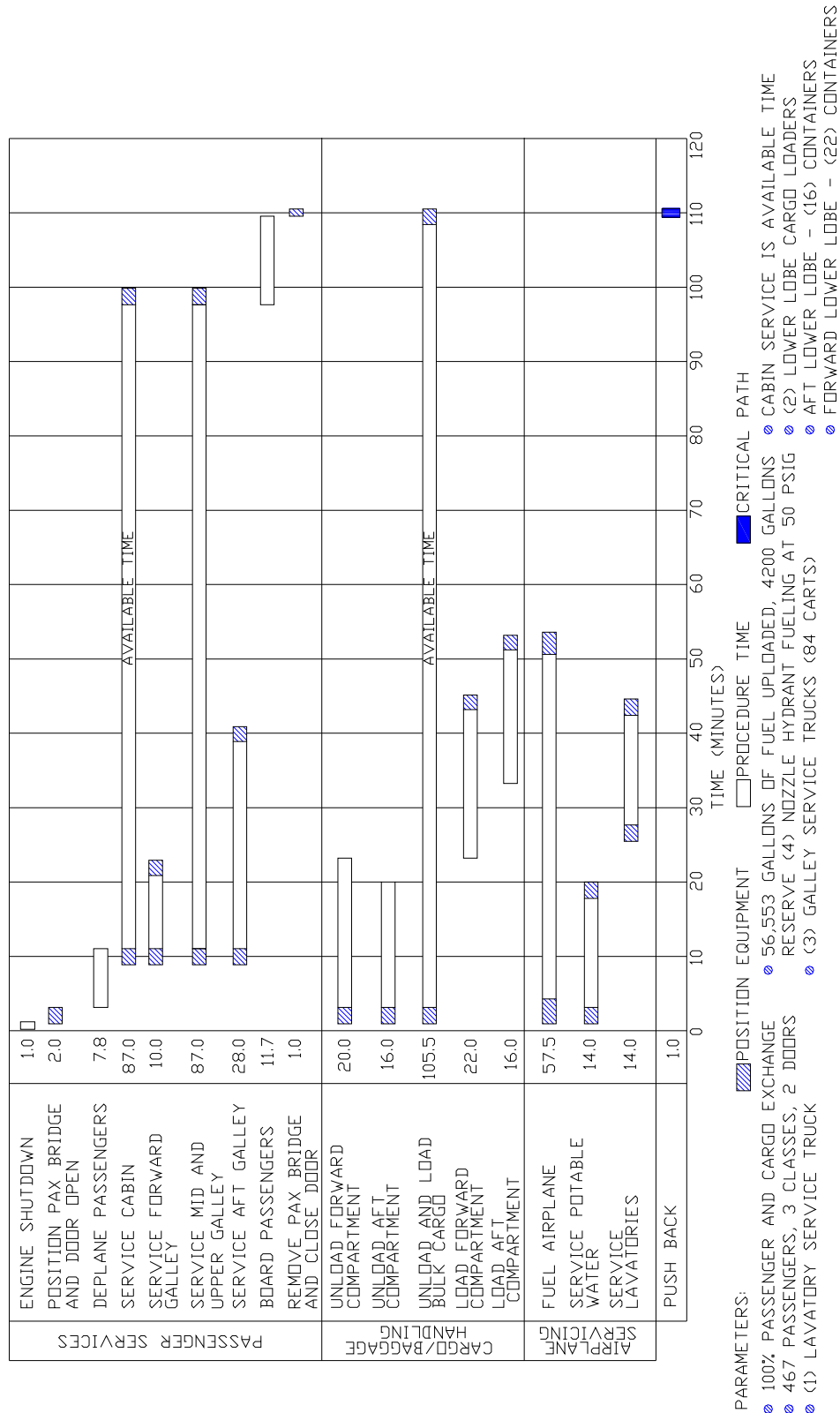
○ POTABLE WATER, AIR CONDITIONING, OR GROUND POWER MAY BE SUPPLIED FROM THE PASSENGER BRIDGE IF SO EQUIPPED

5.1.1 AIRPLANE SERVICING ARRANGEMENT - TYPICAL TURNAROUND MODEL 747-8

60 SEPTEMBER 2008

PRELIMINARY

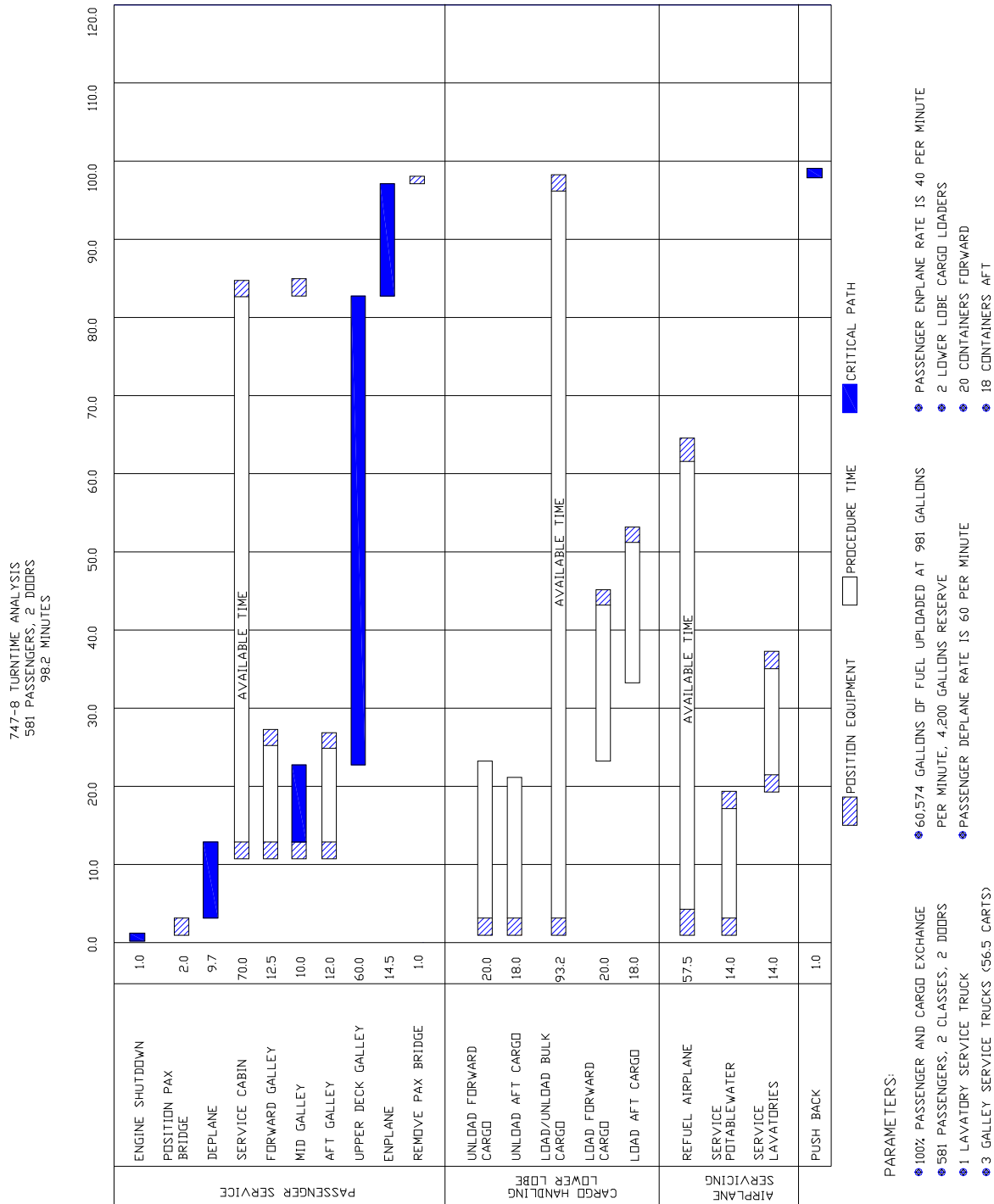
747-8 TURNTIME ANALYSIS
467 PASSENGERS, 2 DOORS
110.5 MINUTES



5.2.1 TERMINAL OPERATIONS - TURNAROUND STATION – ALL PASSENGER MODEL 747-8

D6-58326-3

PRELIMINARY

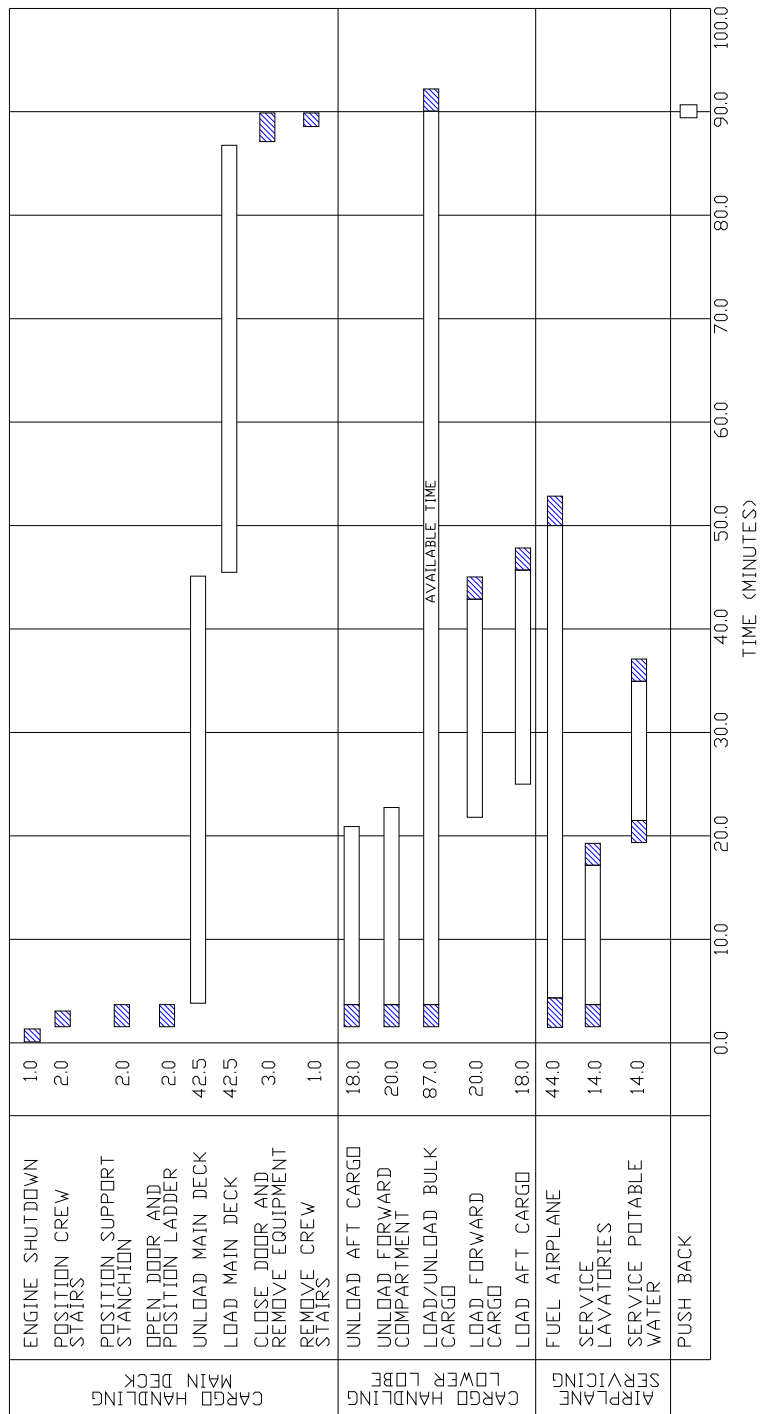


5.2.2 TERMINAL OPERATIONS - TURNAROUND STATION – PASSENGER MODEL 747-8

D6-58326-3

PRELIMINARY

747-8F LOADED USING NOSE CARGO DOOR
90 MINUTES

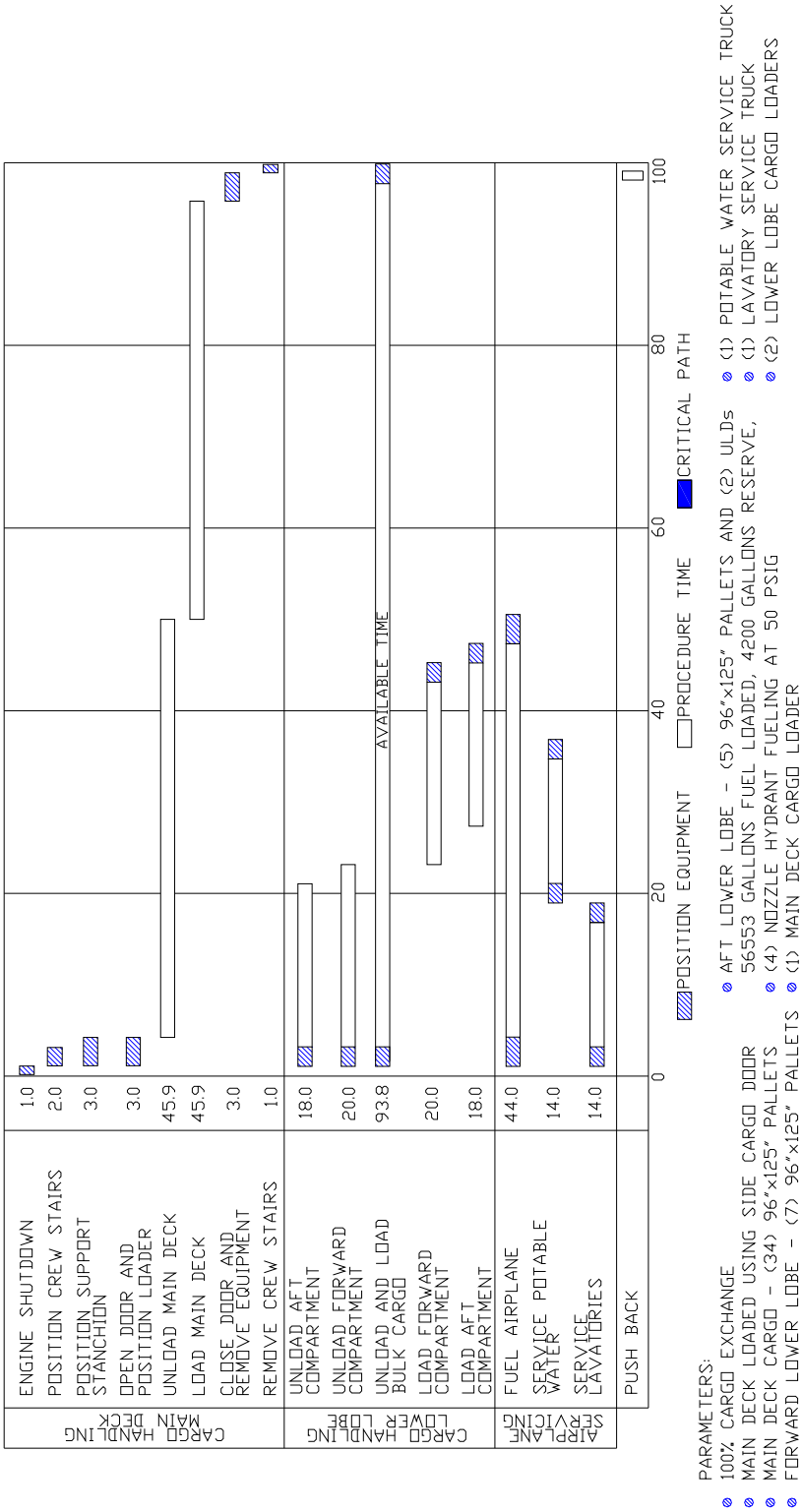


- PARAMETERS:
- 100% CARGO EXCHANGE
 - MAIN DECK LOADED USING NOSE CARGO DOOR
 - MAIN DECK CARGO - (34) 96"x125" PALLETS
 - (1) MAIN DECK CARGO LOADER
 - AFT LOWER LOBE - (5) 96"x125" PALLETS AND (2) ULDS
 - 56,553 GALLONS FUEL LOADED, 4200 GALLONS RESERVE, (4) NOZZLE HYDRANT FUELING AT 50 PSIG
 - FORWARD LOWER LOBE - (7) 96"x125" PALLETS
 - LOWER LOBE CARGO LOADERS
 - POTABLE WATER SERVICE TRUCKS
 - LAVATORY SERVICE TRUCK

5.2.3 TERMINAL OPERATIONS - TURNAROUND STATION - ALL CARGO, NOSE DOOR LOADING MODEL 747-8F

PRELIMINARY

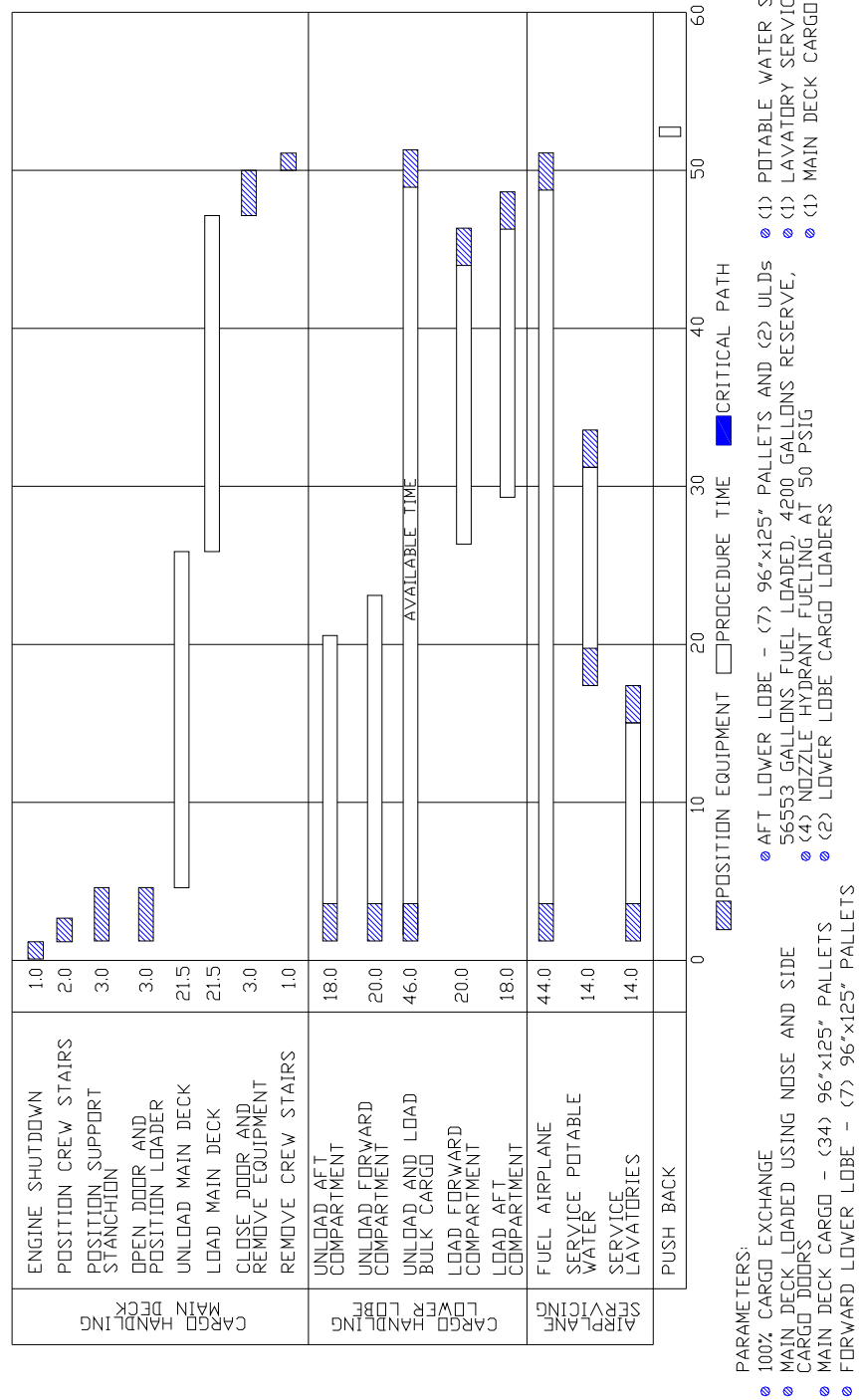
747-8F TURNTIME ANALYSIS
USING SIDE CARGO DOOR
99 MINUTES



5.2.4 TERMINAL OPERATIONS – TURNAROUND STATION – ALL CARGO, SIDE DOOR LOADING
MODEL 747-8F

PRELIMINARY

747-8F TURNTIME ANALYSIS
USING NOSE AND SIDE
CARGO DOORS
52 MINUTES

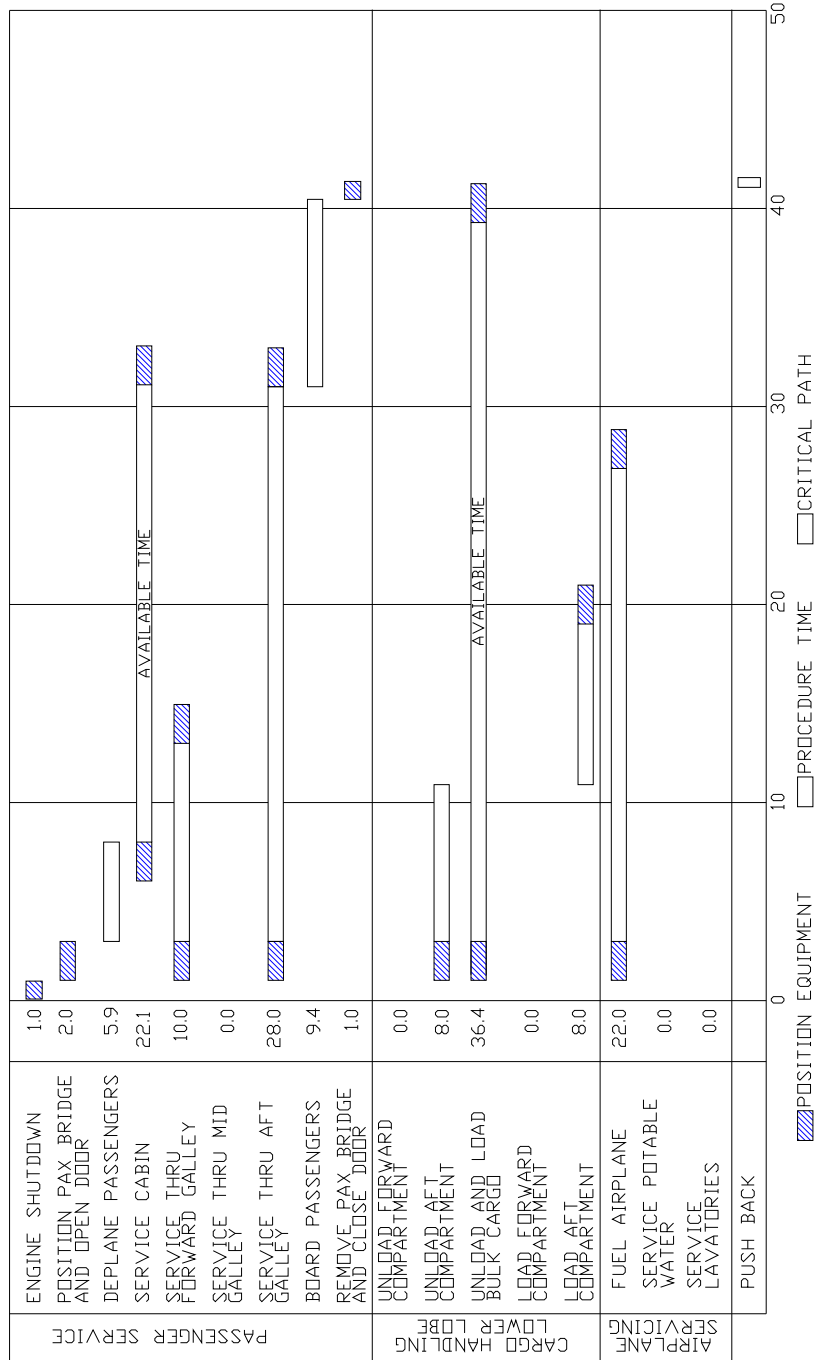


5.2.5 TERMINAL OPERATIONS – TURNAROUND STATION – ALL CARGO, NOSE AND SIDE LOADING

MODEL 747-8F

PRELIMINARY

747-8 TURNTIME ANALYSIS THRU STOP 41.5 MINUTES



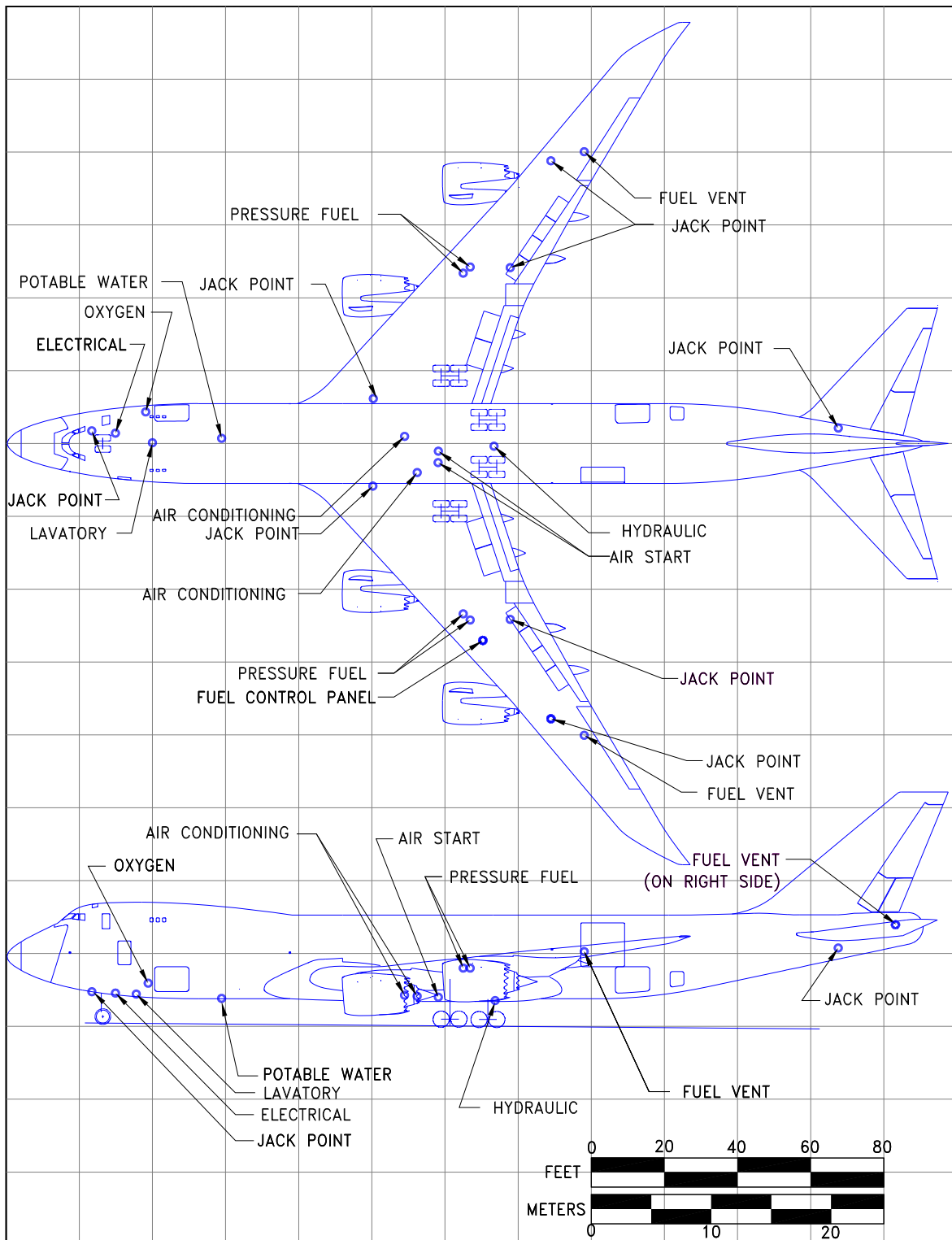
- PARAMETERS:
- 50% PASSENGER AND CARGO EXCHANGE
 - 234 PASSENGERS EXCHANGED AT DOOR L1
 - (1) LAVATORY SERVICE TRUCK
 - 20000 GALLONS FUEL LOADED
 - (2) NOZZLE HYDRANT FUELING AT 50 PSIG
 - CABIN SERVICE IS AVAILABLE TIME
 - AFT LOWER LOBE - (8) CONTAINERS EXCHANGED
 - (2) GALLEY SERVICE TRUCKS (38 CARTS)
 - (1) LOWER LOBE CARGO LOADER

5.3.1 TERMINAL OPERATIONS - EN ROUTE STATION - ALL PASSENGER

MODEL 747-8

D6-58326-3

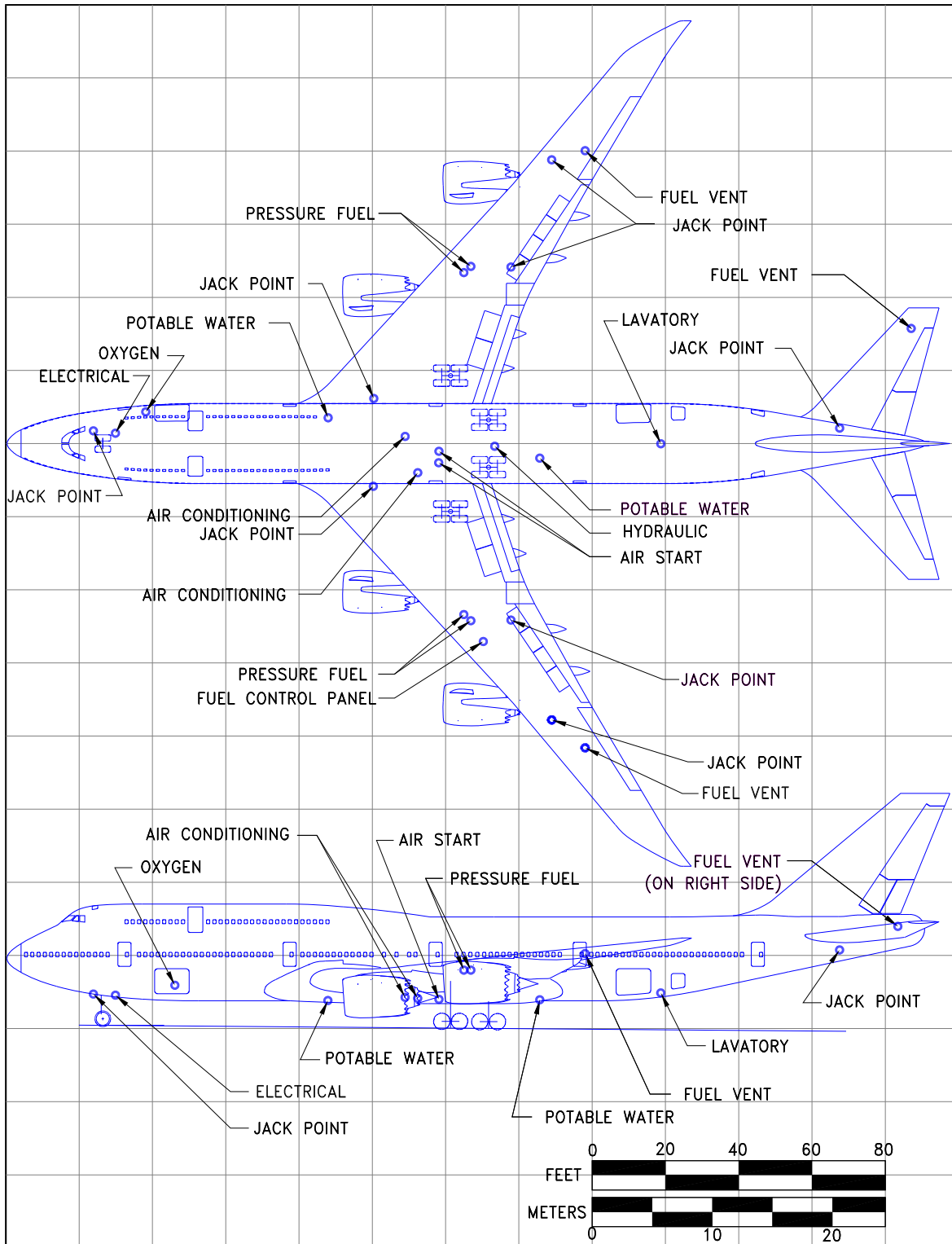
PRELIMINARY



5.4.1 GROUND SERVICE CONNECTIONS

MODEL 747-8F

PRELIMINARY



5.4.2 GROUND SERVICE CONNECTIONS

MODEL 747-8

D6-58326-3

PRELIMINARY

SYSTEM	DISTANCE AFT OF NOSE		DISTANCE FROM AIRPLANE CENTERLINE				HEIGHT ABOVE GROUND			
			LH SIDE		RH SIDE		MINIMUM		MAXIMUM	
	FT-IN	M	FT-IN	M	FT-IN	M	FT-IN	M	FT-IN	M
ELECTRICAL TWO CO-LOCATED CONNECTORS - 90 KVA , 200/115 V AC 400 HZ, 3-PHASE EA.	26 - 9	8.15	-	-	3 - 4	1.02	8 - 1	2.46	9 - 3	2.82
FUEL OUTBOARD OVER- WING PRESSURE CONNECTORS (2 ON EACH WING)	118 - 9	35.94	47 - 8	14.53	47 - 8	14.53	15 - 4	4.67	16 - 0	4.88
INBOARD OVER WING PRESSURE CONNECTORS (2 ON EACH WING)	119 - 9	36.25	46 - 8	14.22	46 - 8	14.22	15 - 3	4.64	15 - 10	4.83
MAX FUELING RATE 500 US GPM (1890 LPM) PER NOZZLE OR 2000 US GPM (7570 LPM)										
TOTAL MAX FUEL PRESSURE 50 PSIG (3.52 KG/CM ²)										
WING FUEL VENT	166 - 4	50.7	92 - 7	28.22	92 - 7	28.22	16 - 10	5.12	19 - 3	5.86
TAIL FUEL VENT										
TANK CAPACITIES:										

5.4.3 GROUND SERVICE CONNECTIONS

MODEL 747-8, 747-8F

PRELIMINARY

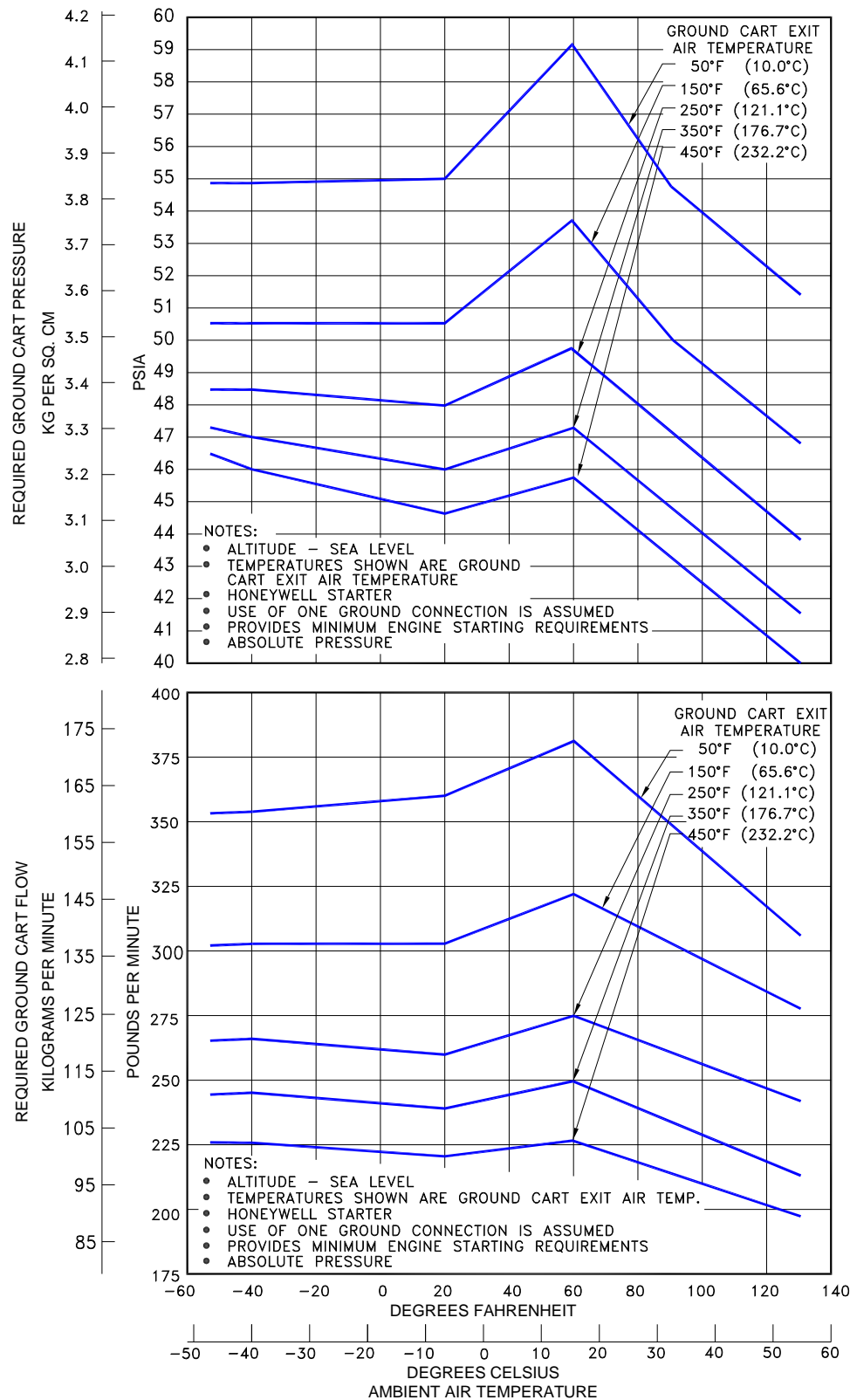
SYSTEM	DISTANCE AFT OF NOSE		DISTANCE FROM AIRPLANE CENTERLINE				HEIGHT ABOVE GROUND			
			LH SIDE		RH SIDE		MINIMUM		MAXIMUM	
	FT-IN	M	FT-IN	M	FT-IN	M	FT-IN	M	FT-IN	M
LAVATORY ONE SERVICE PANEL THREE SERVICE CONNECTIONS DRAIN: ONE 4-IN (10.2 CM) FLUSH: TWO 1-IN (2.5 CM) FLUSH REQUIREMENTS FLOW: 10 GPM (38 LPM) PRESSURE: 30 PSIG (2.11 KG/CM ²) TOTAL CAPACITY, 4 TANKS 300 US GAL (1140 L)	178 - 4	54.36	0	0	0	0	8 - 8	2.64	9 - 8	2.95
PNEUMATIC TWO 3-IN(7.6CM) HIGH- PRESSURE PORTS TWO 8-IN (0.20 M) GROUND CONDITIONED AIR CONNECTIONS	109 -10	33.48	2 - 0	0.61	-	-	6 - 7	2.01	7 - 5	2.27
	109 -10	33.48	3 - 0	0.91	-	-	6 - 7	2.01	7 - 5	2.27
	102 - 4	31.19	-	-	1 - 10	0.55	7 - 4	2.24	8 - 3	2.52
	106 - 8	32.51	7 - 11	2.40	-	-	7 - 0	2.13	7 - 11	2.40
POTABLE WATER ONE SERVICE CONNECTION CONNECTOR SIZE 3/4 IN (1.95 CM) TANK CAPACITY - 330 U.S GAL (1,250 L) FILL PRESSURE - 30 PSIG (2.11 KG/SQ CM) FILL RATE - 30 GPM (113.5 LPM) DRAIN SIZE 1 IN (2.54 CM) SECOND CONNECTION (ON 747-400ER ONLY)	87 - 8	26.72	-	-	1 - 4	0.5	7 - 4	2.24	8 - 1	2.46
	145 - 6	44.35	2 - 10	0.87	-	-	7 - 3	2.21	8 - 0	2.44
HYDRAULIC ONE SERVICE PANEL 4 RESERVOIRS ENG 1 - 9.5 U.S. GAL (35.9 L) ENG 2 - 5.5 U.S. GAL (20.8 L) ENG 3 - 5.5 U.S. GAL (20.8 L) ENG 4 - 9.5 U.S. GAL (35.9 L) 150 PSI (10.6 KG/CM ²) MAX	127 - 4	38.81	0 - 10	0.25	-	-	7 - 0	2.13	7 - 0	2.13
OXYGEN ONE SERVICE CONNECTION - SIZE 3/16 IN (0.48 CM) 1850 PSIG (130 KG/CM ²) MAX	39 - 2	11.94	-	-	8 - 4	2.54	13 - 7	4.14	14 - 8	4.47

5.4.4 GROUND SERVICING CONNECTIONS

D6-58326-3

PRELIMINARY

MODEL 747-8, 747-8F

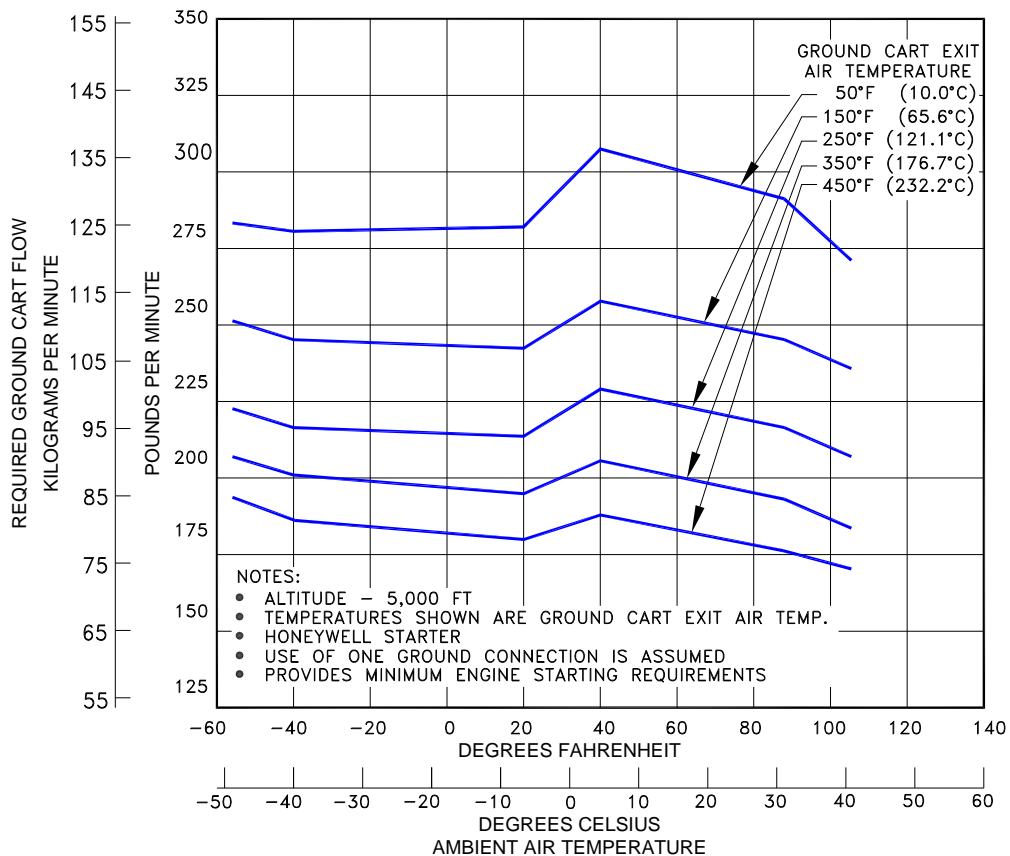
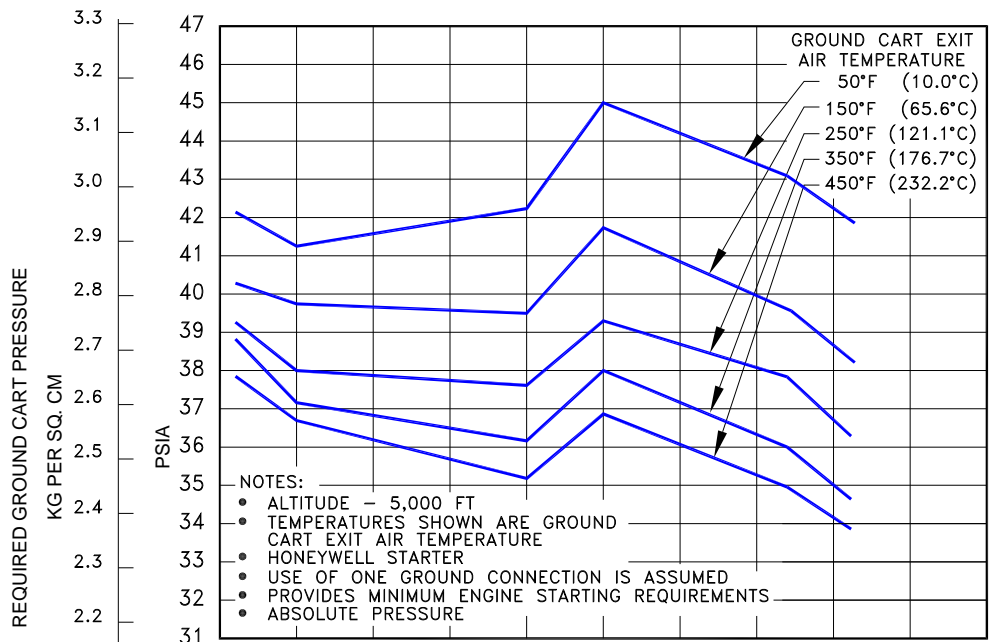


5.5.1 ENGINE START PNEUMATIC REQUIREMENTS - SEA LEVEL

MODEL 747-8, 747-8F

D6-58326-3

PRELIMINARY

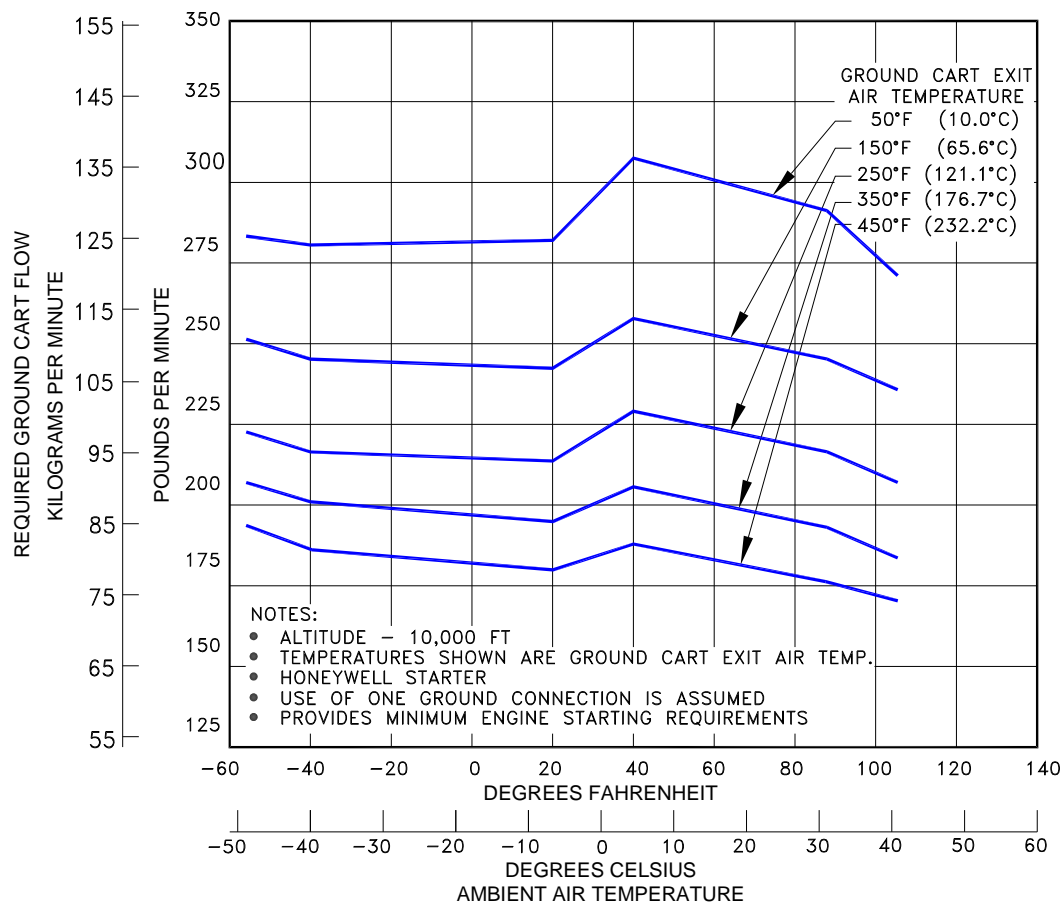
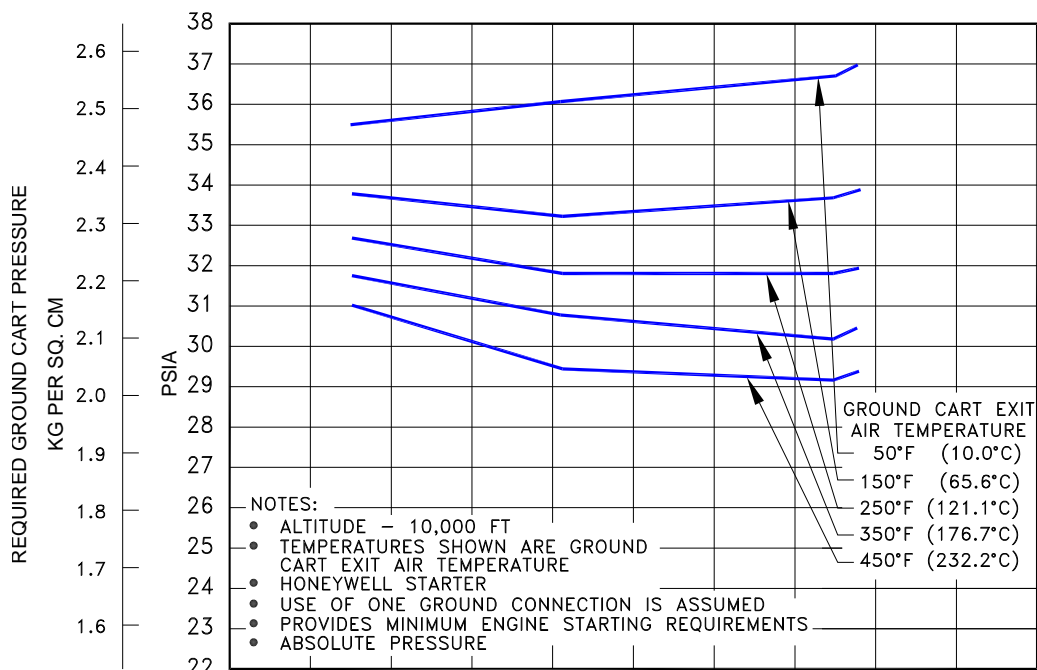


5.5.2 ENGINE START PNEUMATIC REQUIREMENTS - 5,000 FT

MODEL 747-8, 747-8F

D6-58326-3

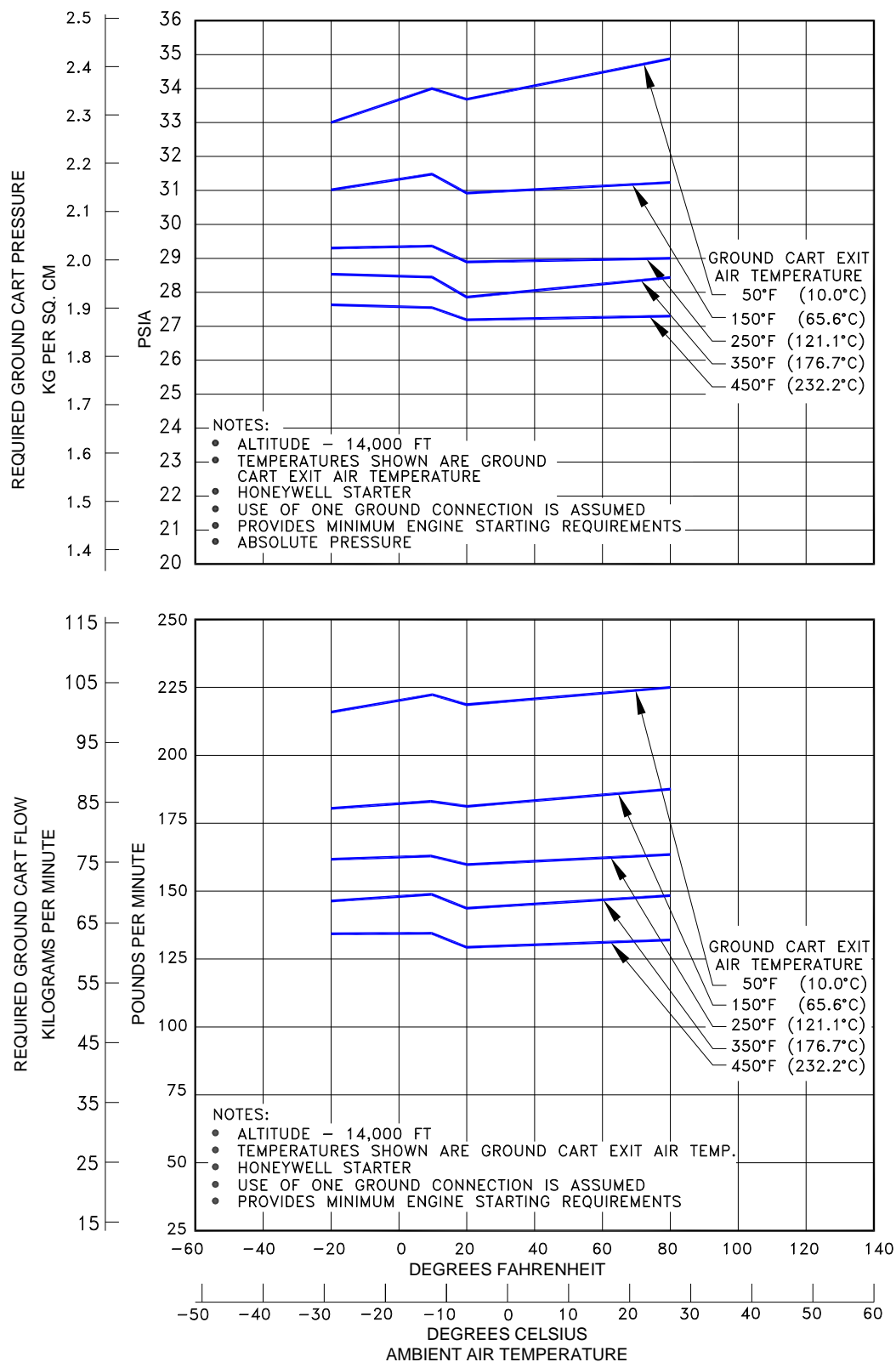
PRELIMINARY



5.5.3 ENGINE START PNEUMATIC REQUIREMENTS - 10,000 FT

MODEL 747-8, 747-8F

PRELIMINARY

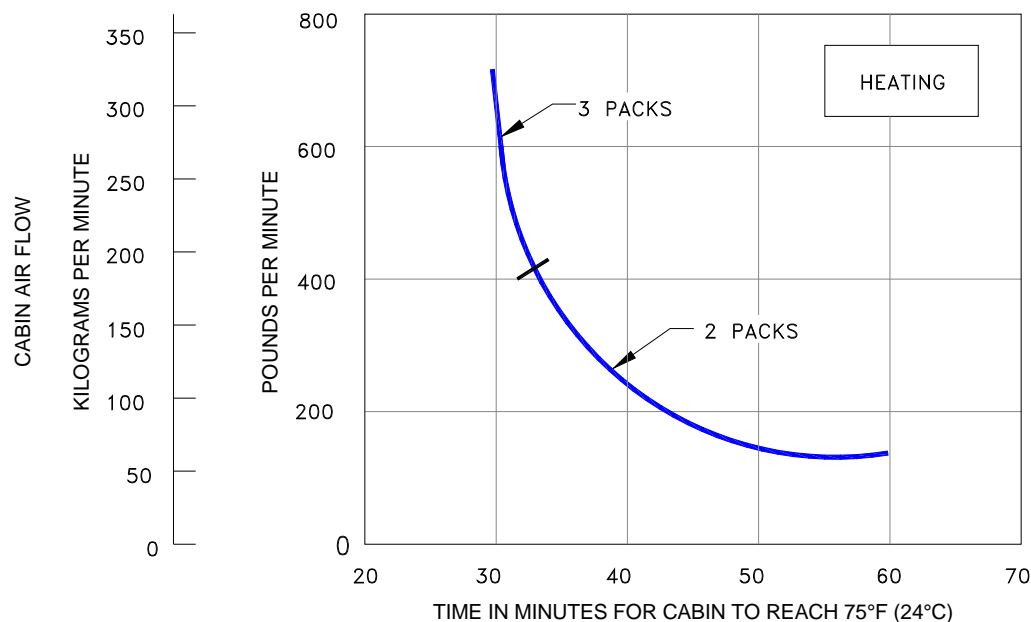


5.5.4 ENGINE START PNEUMATIC REQUIREMENTS – 14,000 FT

MODEL 747-8, 747-8F

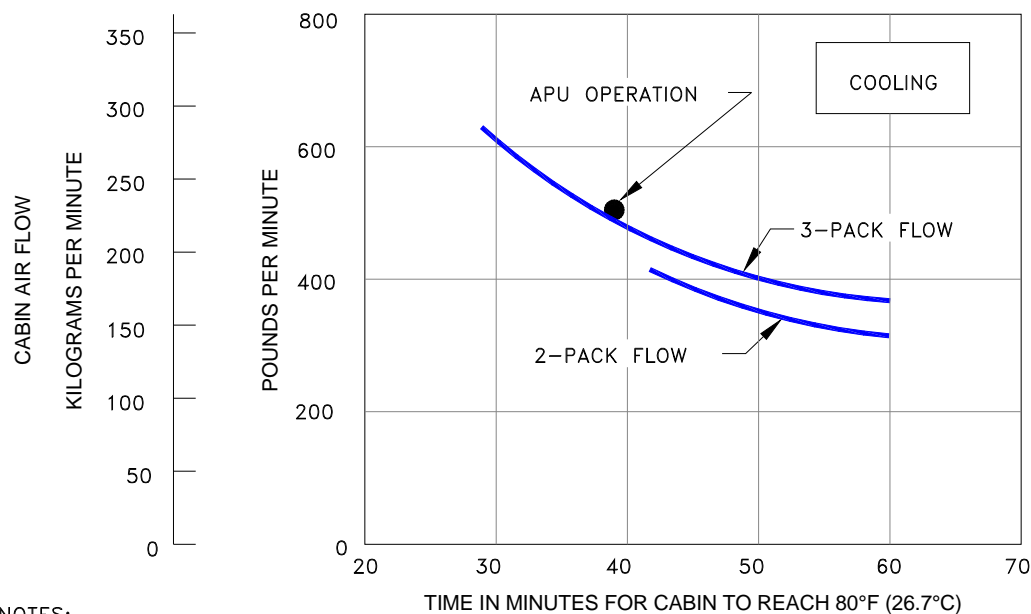
D6-58326-3

PRELIMINARY



NOTES:

- AMBIENT AND INITIAL CABIN TEMP. -25°F (-32°C)
- PACK OUTLET TEMP. 160°F (71°C)
- RECIRCULATING FANS OFF
- NO PASSENGERS
- DOORS CLOSED
- NO GALLEY LOAD
- NO LIGHTING LOAD



NOTES:

- INITIAL CABIN TEMP. AT 115°F (46.1°C)
- OUTSIDE AIR TEMP. AT 100°F (37.8°C)
- TEMP. AT GROUND CONNECTION 400°F (204.4°C)
- NO PASSENGERS
- DOORS CLOSED
- NO GALLEY LOAD
- NO LIGHTING LOAD

5.6.1 GROUND PNEUMATIC POWER REQUIREMENTS - HEATING/COOLING

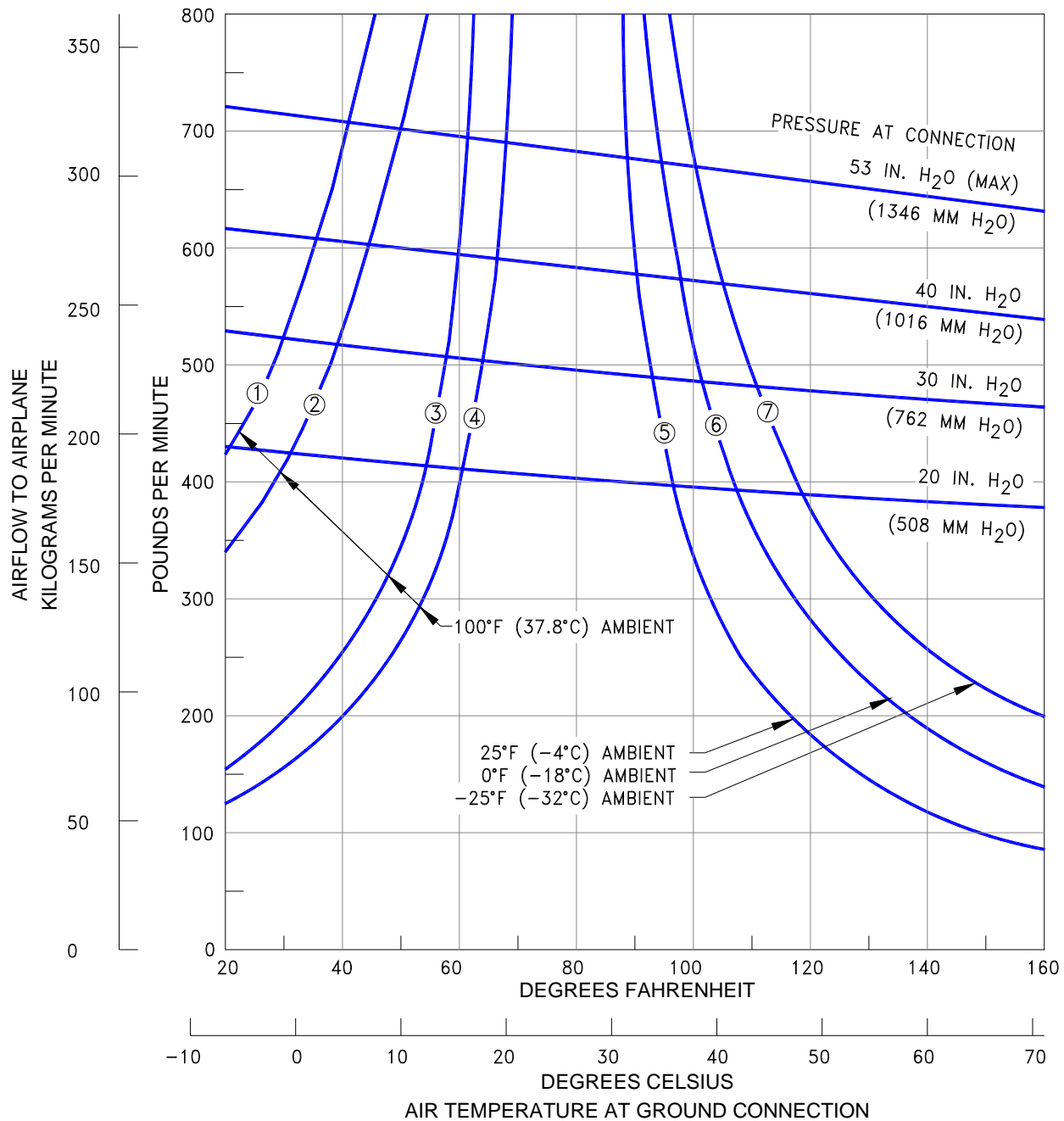
MODEL 747-8, 747-8F (PRELIMINARY DATA AND WILL BE UPDATED AT A LATER DATE)

PRELIMINARY

CONDITIONS:

ALL DOORS AND HATCHES CLOSED

- ① 75°F (23.9°C) CABIN TEMP. 590 OCCUPANTS: 28,000 BTU/HR (7,050 KCAL/HR)
SOLAR LOAD AND 75,000 BTU/HR (18,900 KCAL/HR) ELECTRICAL LOAD
- ② 80°F (26.7°C) CABIN TEMP. HEAT LOADS SAME AS ①, ABOVE
- ③ 75°F (23.9°C) CABIN TEMP. 3 OCCUPANTS 28,000 BTU/HR (7,050 KCAL/HR) SOLAR LOAD
- ④ 80°F (26.7°C) CABIN TEMP. HEAT LOADS SAME AS ③, ABOVE
- ⑤ ⑥ ⑦ 75°F (23.9°C) CABIN TEMP. NO OCCUPANTS OR HEAT LOADS

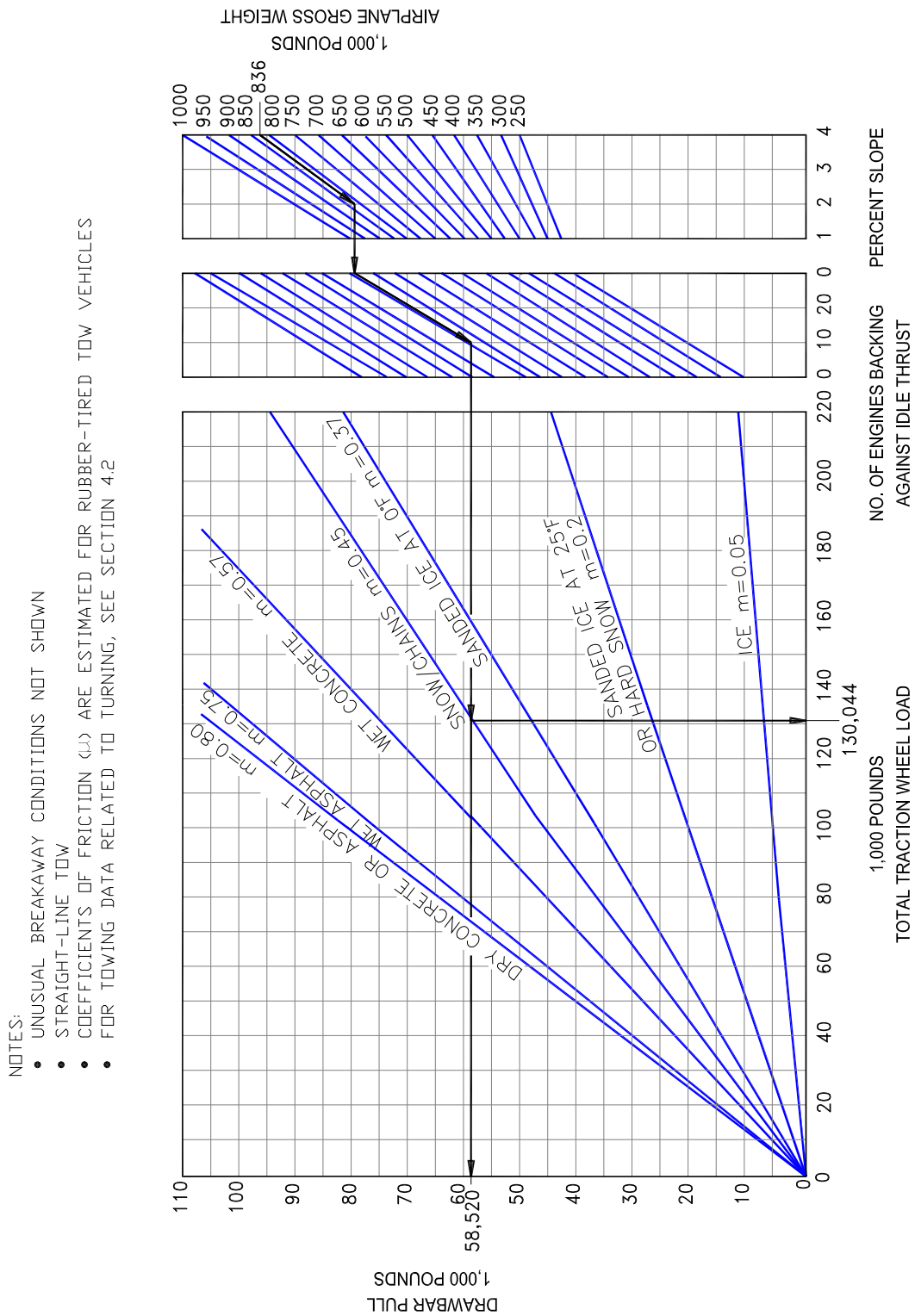


5.7.1 CONDITIONED AIR FLOW REQUIREMENTS

MODEL 747-8, 747-8F (PRELIMINARY DATA AND WILL BE UPDATED AT A LATER DATE)

D6-58326-3

PRELIMINARY



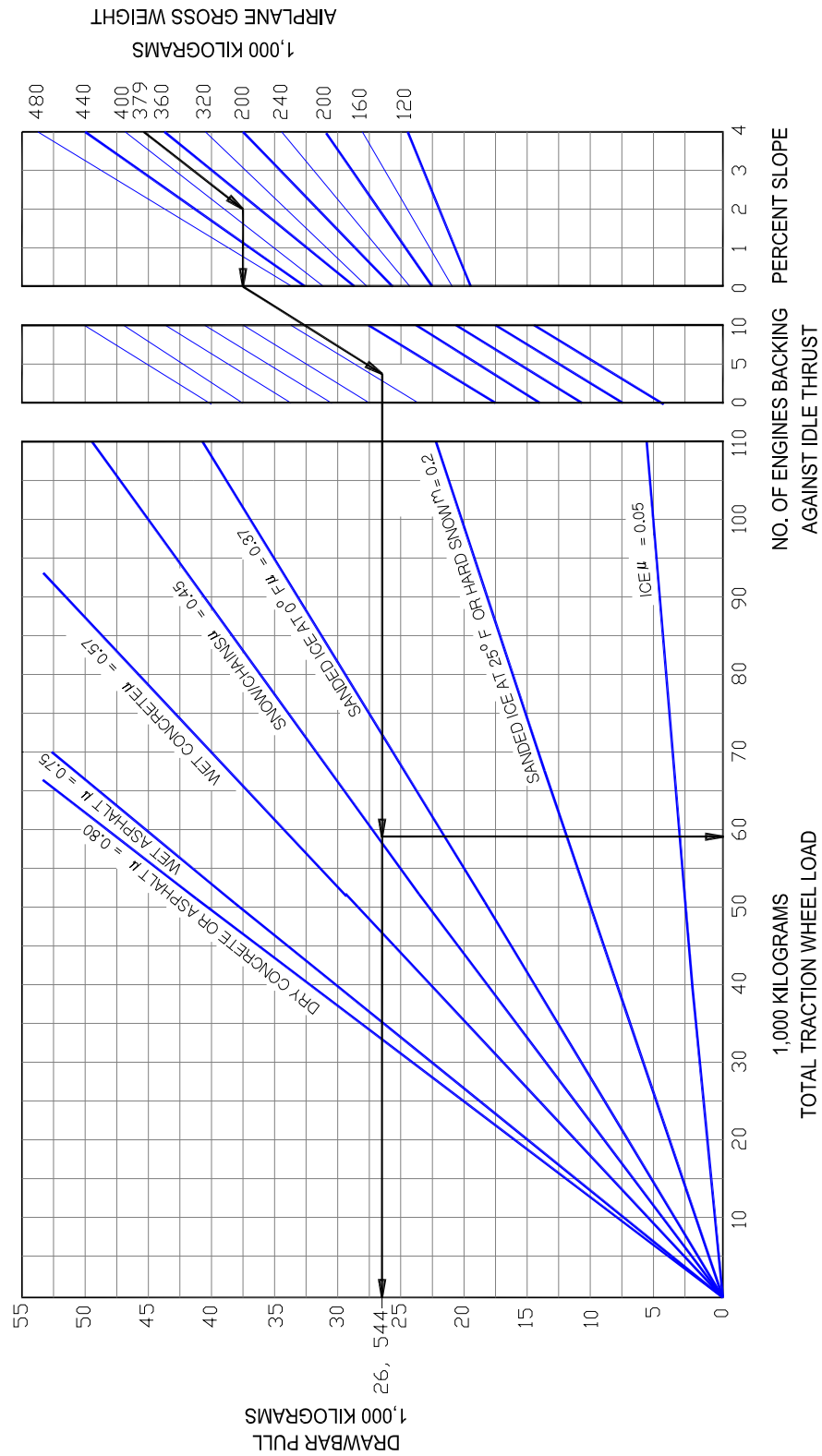
5.8.1 GROUND TOWING REQUIREMENTS - ENGLISH UNITS

MODEL 747-8, 747-8F

PRELIMINARY

NOTES:

- UNUSUAL BREAKAWAY CONDITIONS NOT SHOWN
- STRAIGHT-LINE TOW
- COEFFICIENTS OF FRICTION (μ) ARE ESTIMATED FOR RUBBER-TIRED TOW VEHICLES
- FOR TOWING DATA RELATED TO TURNING, SEE SECTION 4.2



5.8.2 GROUND TOWING REQUIREMENTS - METRIC UNITS

MODEL 747-8, 747-8F

D6-58326-3

PRELIMINARY

6.0 JET ENGINE WAKE AND NOISE DATA

6.1 Jet Engine Exhaust Velocities and Temperatures

6.2 Airport and Community Noise

PRELIMINARY

6.0 JET ENGINE WAKE AND NOISE DATA

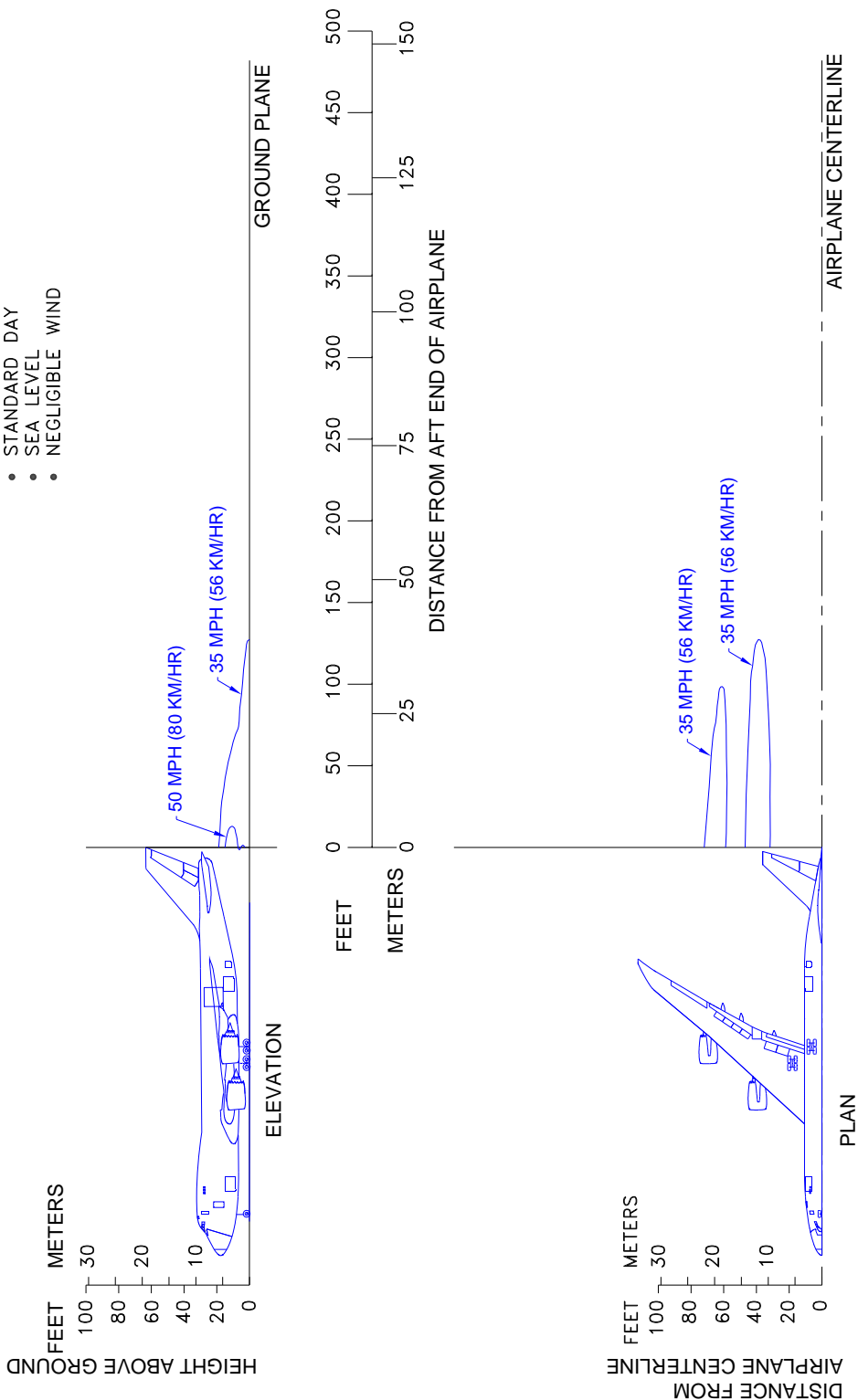
6.1 Jet Engine Exhaust Velocities and Temperatures

This section shows exhaust velocity and temperature contours aft of the 747-8 Intercontinental and 747-8 Freighter airplanes due to the use of the same engine and same weight for both airplanes. The contours were calculated from a standard computer analysis using three-dimensional viscous flow equations with mixing of primary, fan, and free-stream flow. The presence of the ground plane is included in the calculations as well as engine tilt and toe-in. Mixing of flows from the engines is also calculated. The analysis does not include thermal buoyancy effects which tend to elevate the jet wake above the ground plane. The buoyancy effects are considered to be small relative to the exhaust velocity and therefore are not included.

The graphs show jet wake velocity and temperature contours for a representative engine. The results are valid for sea level, static, standard day conditions. The effect of wind on jet wakes was not included. There is evidence to show that a downwind or an upwind component does not simply add or subtract from the jet wake velocity, but rather carries the whole envelope in the direction of the wind. Crosswinds may carry the jet wake contour far to the side at large distances behind the airplane.

PRELIMINARY

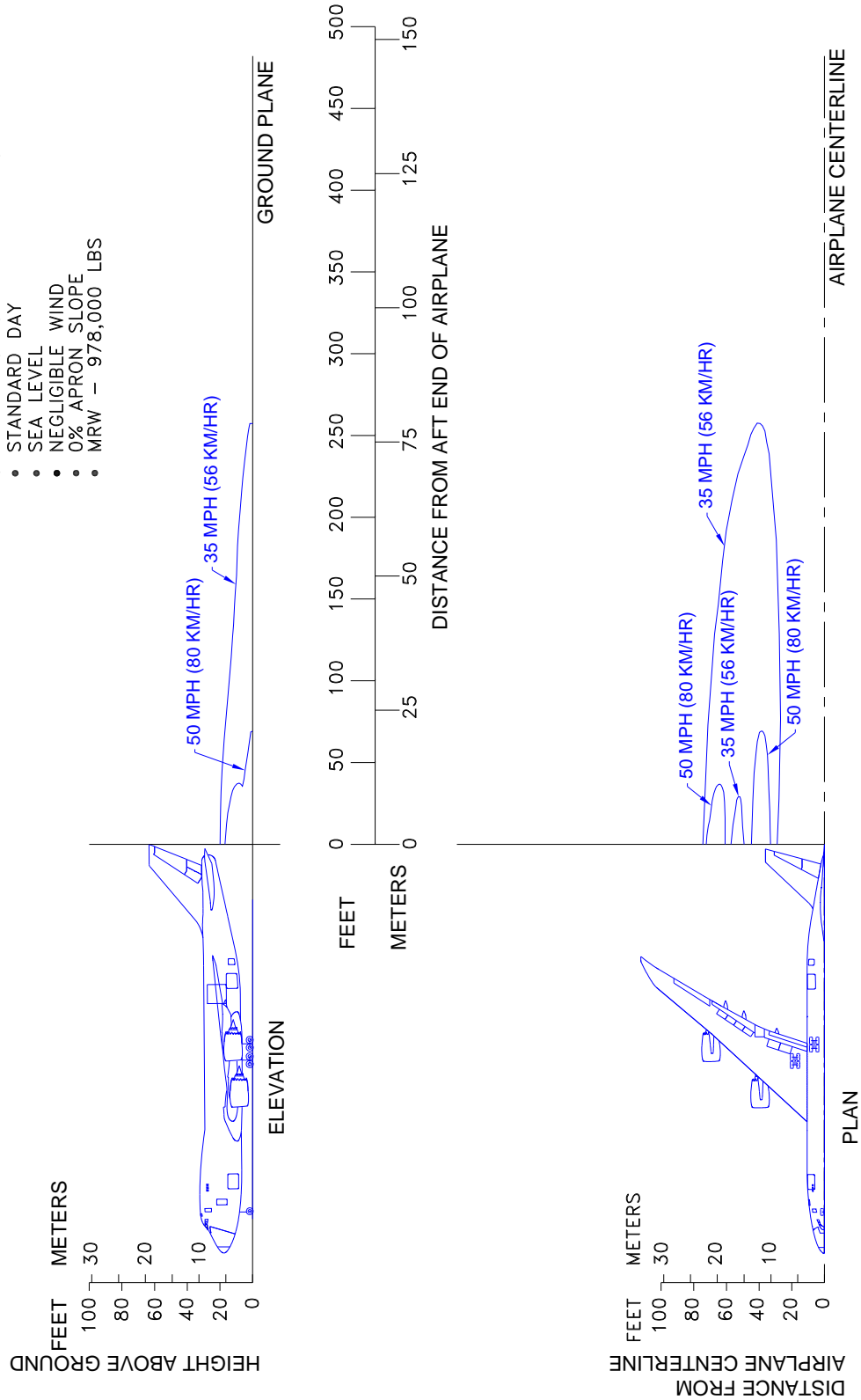
- NOTES:
- ENGINE THRUST (2430 LBS), IDLE SETTINGS
 - CONTOURS CALCULATED FROM COMPUTER DATA
 - STANDARD DAY
 - SEA LEVEL
 - NEGLIGIBLE WIND



6.1.1 JET ENGINE EXHAUST VELOCITY CONTOURS – IDLE THRUST
MODEL 747-8, 747-8F

PRELIMINARY

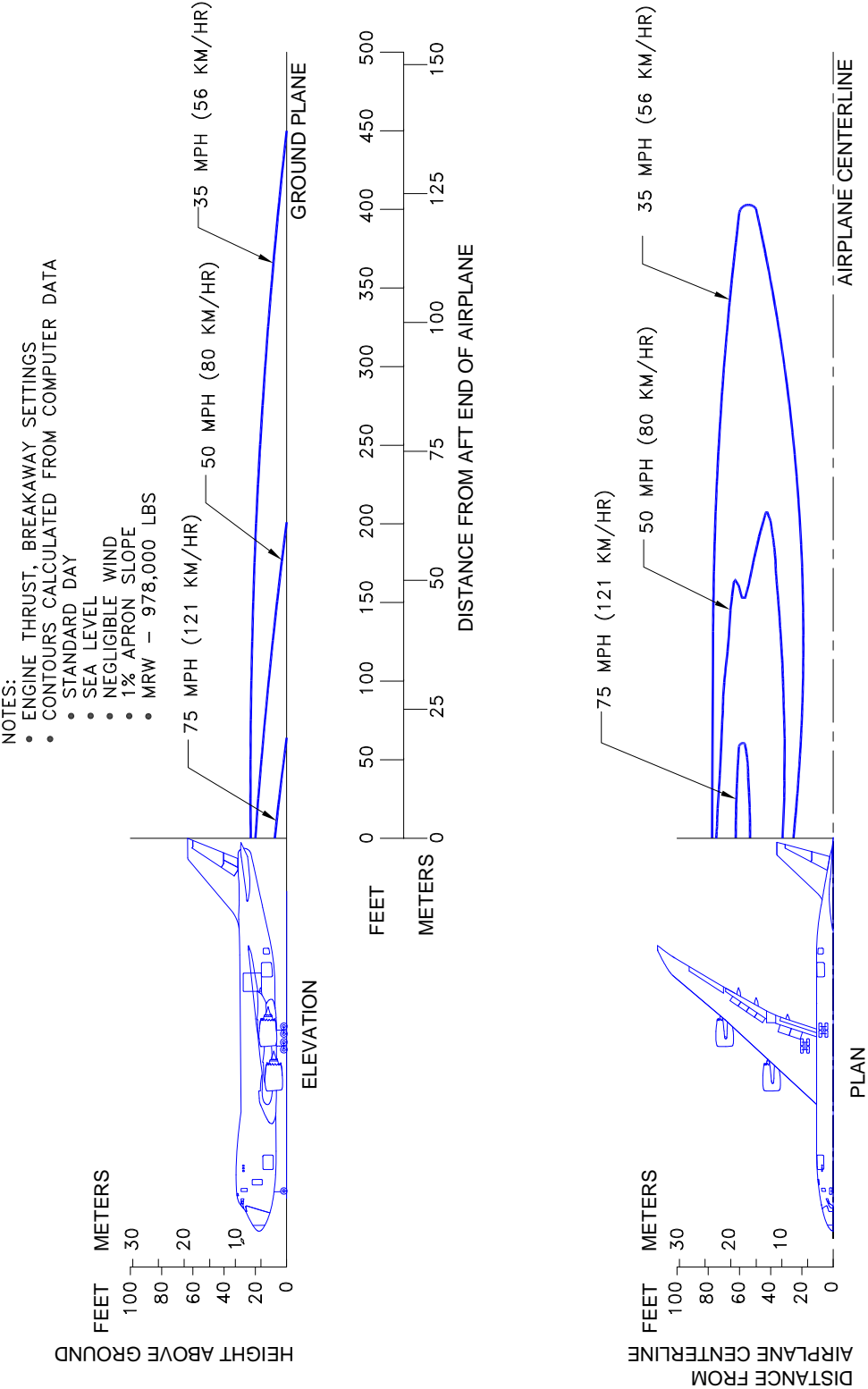
- NOTES:
- ENGINE THRUST, BREAKAWAY SETTINGS
 - CONTOURS CALCULATED FROM COMPUTER DATA
 - STANDARD DAY
 - SEA LEVEL
 - NEGLIGIBLE WIND
 - 0% APRON SLOPE
 - MRW – 978,000 LBS



6.1.2 JET ENGINE EXHAUST VELOCITY CONTOURS – BREAKAWAY THRUST – LEVEL PAVEMENT MODEL 747-8, 747-8F

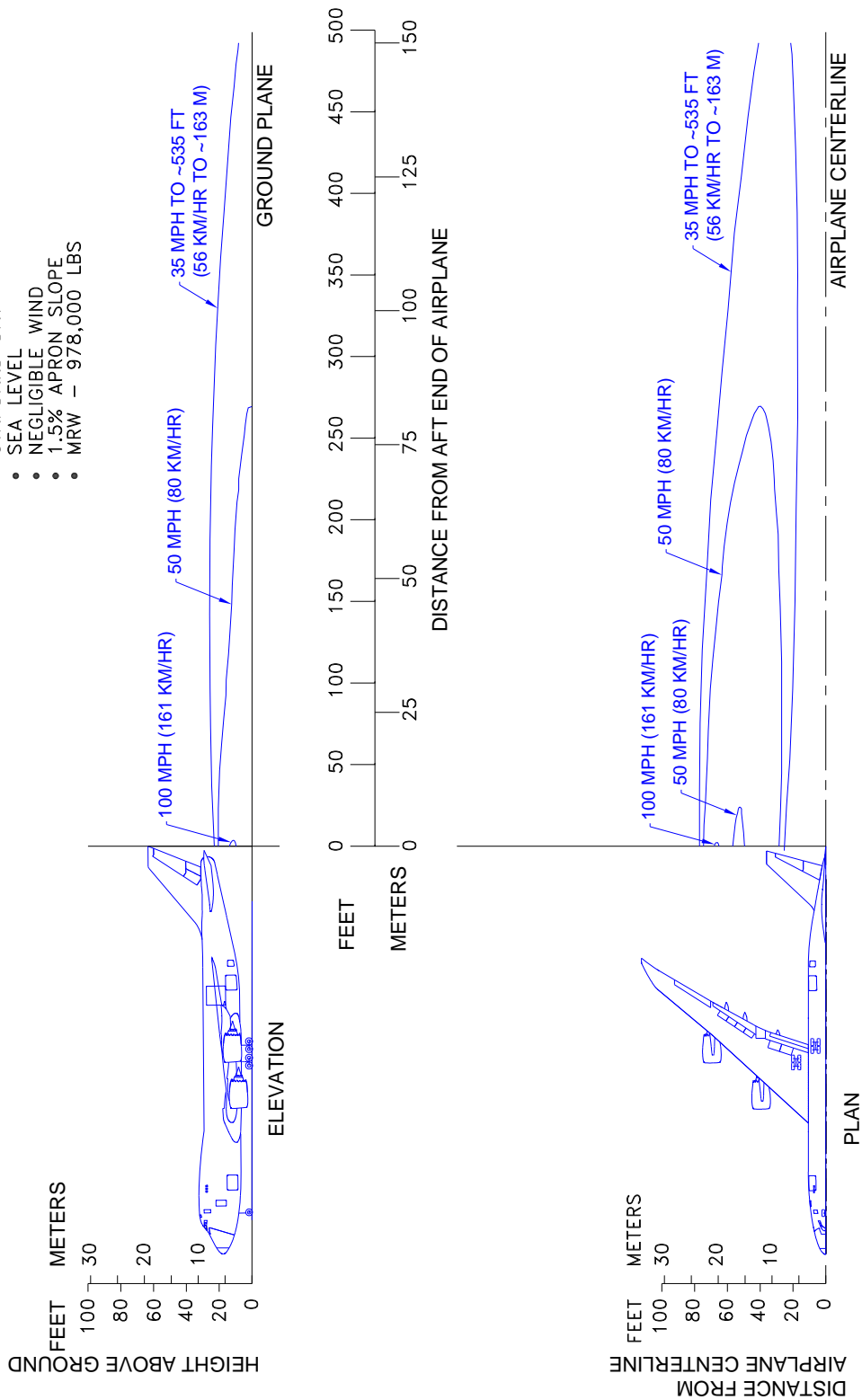
PRELIMINARY

6.1.3 JET ENGINE EXHAUST VELOCITY CONTOURS – BREAKAWAY THRUST -
1% PAVEMENT UPSLOPE
MODEL 747-8, 747-8F



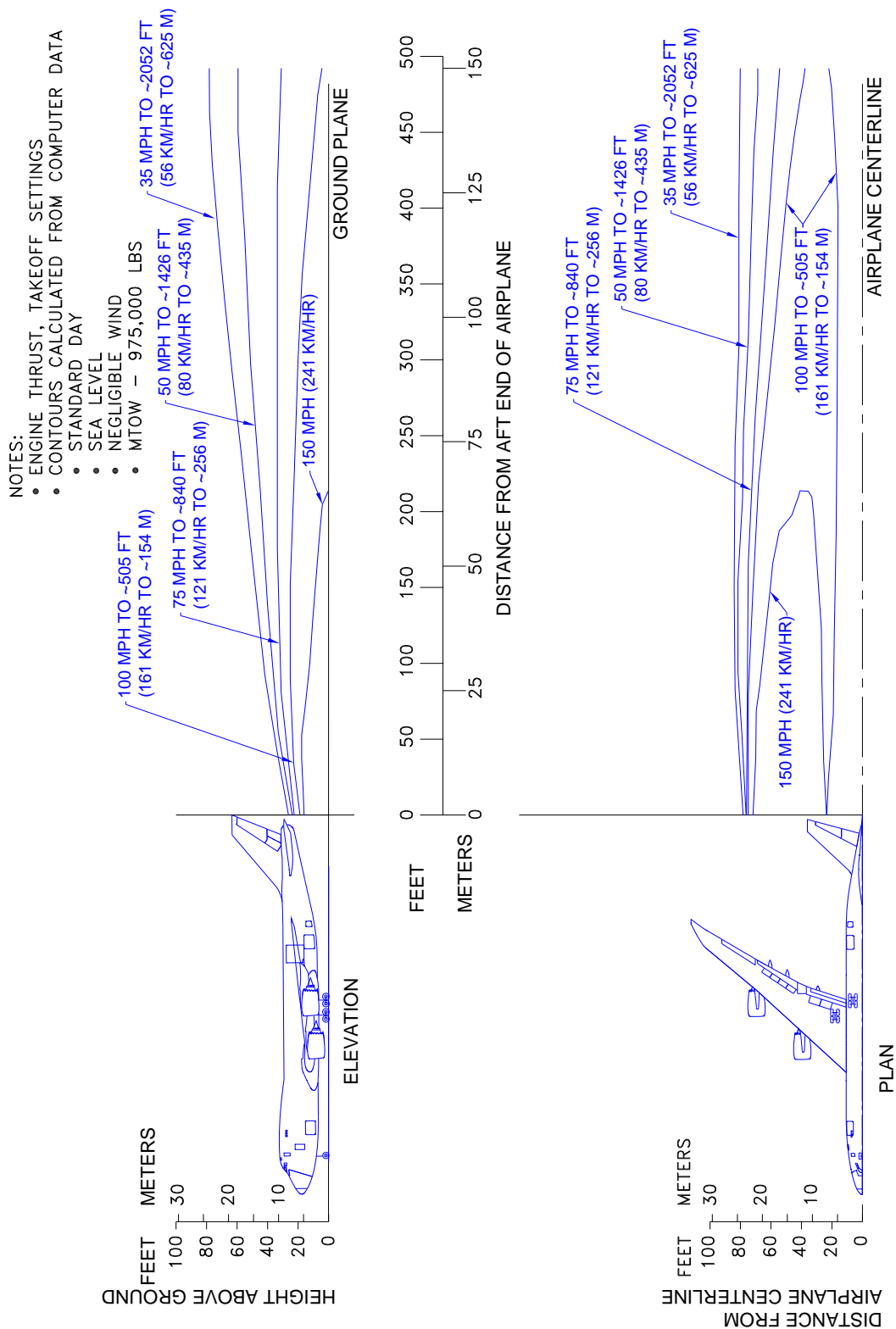
PRELIMINARY

- NOTES:
- ENGINE THRUST, BREAKAWAY SETTINGS
 - CONTOURS CALCULATED FROM COMPUTER DATA
 - STANDARD DAY
 - SEA LEVEL
 - NEGLIGIBLE WIND
 - 1.5% APRON SLOPE
 - MRW - 978,000 LBS



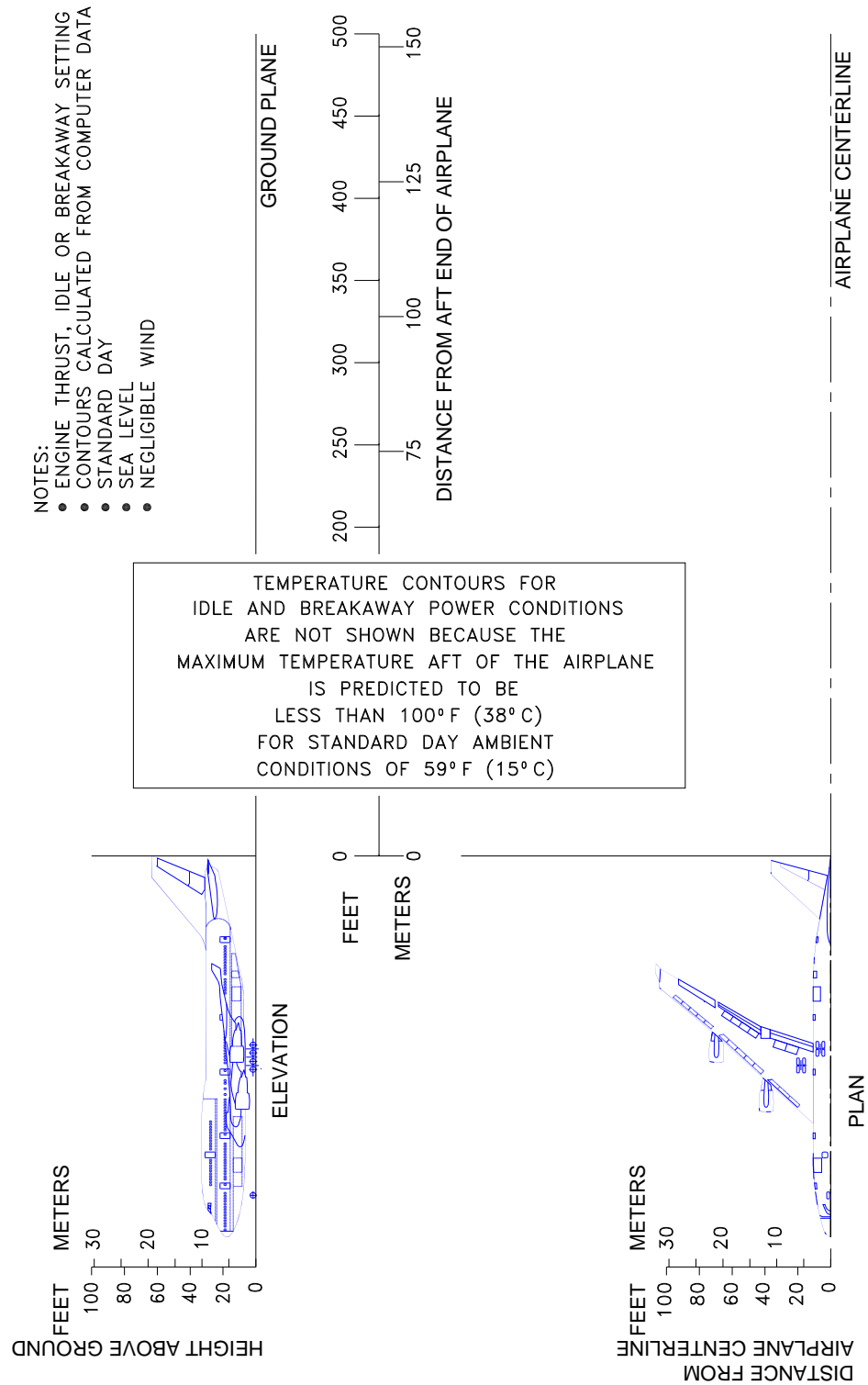
6.1.4 JET ENGINE EXHAUST VELOCITY CONTOURS - BREAKAWAY THRUST - 1.5% PAVEMENT UPSLOPE
MODEL 747-8, 747-8F

PRELIMINARY



6.1.5 JET ENGINE EXHAUST VELOCITY CONTOURS - TAKEOFF THRUST
MODEL 747-8, 747-8F

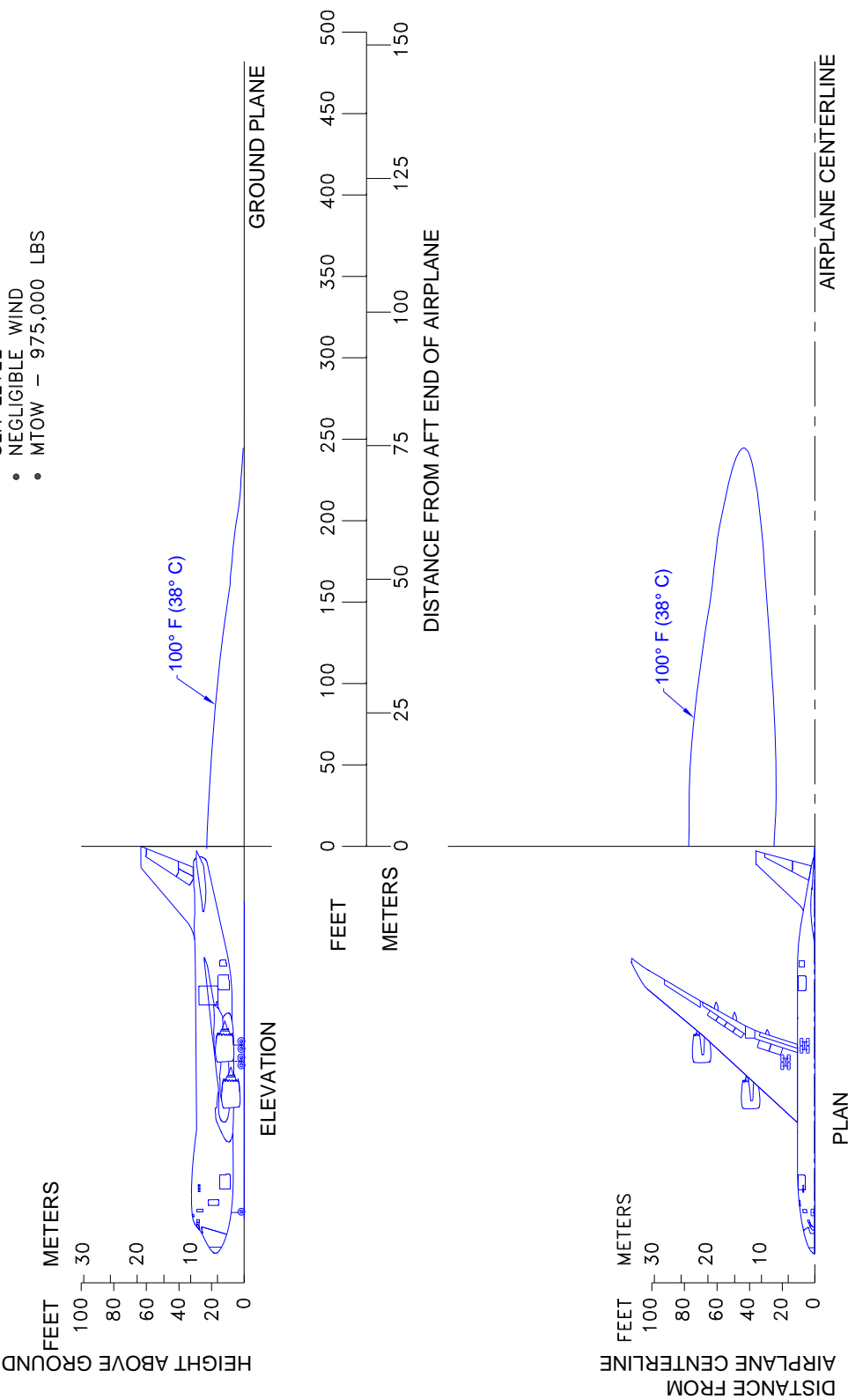
PRELIMINARY



6.1.6 JET ENGINE EXHAUST TEMPERATURE CONTOURS – IDLE AND BREAKAWAY MODEL 747-8, 747-8F

PRELIMINARY

- NOTES:
- ENGINE THRUST, TAKEOFF SETTINGS
 - CONTOURS CALCULATED FROM COMPUTER DATA
 - STANDARD DAY
 - SEA LEVEL
 - NEGLIGIBLE WIND
 - MTOW - 975,000 LBS



6.1.7 JET ENGINE EXHAUST TEMPERATURE CONTOURS - TAKEOFF THRUST
MODEL 747-8, 747-8F

PRELIMINARY

6.2 Airport and Community Noise

Airport noise is of major concern to the airport and community planner. The airport is a major element in the community's transportation system and, as such, is vital to its growth. However, the airport must also be a good neighbor, and this can be accomplished only with proper planning. Since aircraft noise extends beyond the boundaries of the airport, it is vital to consider the impact on surrounding communities. Many means have been devised to provide the planner with a tool to estimate the impact of airport operations. Too often they oversimplify noise to the point where the results become erroneous. Noise is not a simple subject; therefore, there are no simple answers.

The cumulative noise contour is an effective tool. However, care must be exercised to ensure that the contours, used correctly, estimate the noise resulting from aircraft operations conducted at an airport.

The size and shape of the single-event contours, which are inputs into the cumulative noise contours, are dependent upon numerous factors. They include the following:

1. Operational Factors

- (a) Aircraft Weight-Aircraft weight is dependent on distance to be traveled, en route winds, payload, and anticipated aircraft delay upon reaching the destination.
- (b) Engine Power Settings-The rates of ascent and descent and the noise levels emitted at the source are influenced by the power setting used.
- (c) Airport Altitude-Higher airport altitude will affect engine performance and thus can influence noise.

2. Atmospheric Conditions-Sound Propagation

- (a) Wind-With stronger headwinds, the aircraft can take off and climb more rapidly relative to the ground. Also, winds can influence the distribution of noise in surrounding communities.
- (b) Temperature and Relative Humidity-The absorption of noise in the atmosphere along the transmission path between the aircraft and the ground observer varies with both temperature and relative humidity.

PRELIMINARY

3. Surface Condition-Shielding, Extra Ground Attenuation (EGA)

- (a) Terrain-If the ground slopes down after takeoff or up before landing, noise will be reduced since the aircraft will be at a higher altitude above ground. Additionally, hills, shrubs, trees, and large buildings can act as sound buffers.

All these factors can alter the shape and size of the contours appreciable. To demonstrate the effect of some of these factors, estimated noise level contours for two different operating conditions are shown below. These contours reflect a given noise level upon a ground level plane at runway elevation.

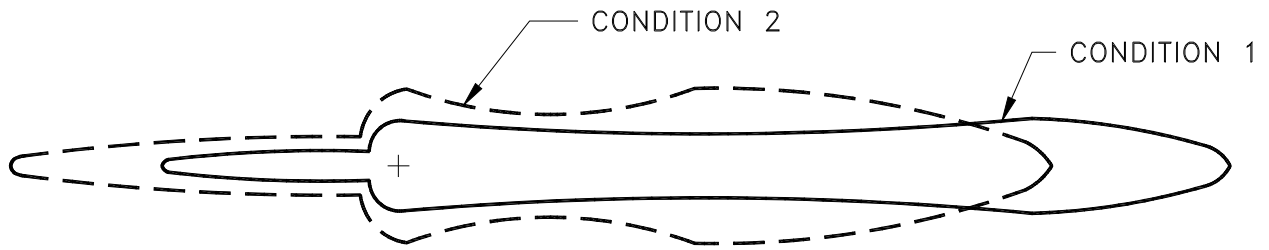
Condition 1

Landing

Maximum Structural Landing
Weight
10-knot Headwind
3^o Approach
84 °F
Humidity 15%

Takeoff

Maximum Gross Takeoff Weight
Zero Wind
84 °F
Humidity 15%



Condition 2

Landing:

85% of Maximum Structural
Landing Weight
10-knot Headwind
3^o Approach
59 °F
Humidity 70%

Takeoff:

80% of Maximum Gross Takeoff
Weight
10-knot Headwind
59 °F
Humidity 70%

PRELIMINARY

As indicated from these data, the contour size varies substantially with operating and atmospheric conditions. Most aircraft operations are, of course, conducted at less than maximum gross weights because average flight distances are much shorter than maximum aircraft range capability and average load factors are less than 100%. Therefore, in developing cumulative contours for planning purposes, it is recommended that the airlines serving a particular city be contacted to provide operational information.

In addition, there are no universally accepted methods for developing aircraft noise contours or for relating the acceptability of specific zones to specific land uses. It is therefore expected that noise contour data for particular aircraft and the impact assessment methodology will be changing. To ensure that the best currently available information of this type is used in any planning study, it is recommended that it be obtained directly from the Office of Environmental Quality in the Federal Aviation Administration in Washington, D.C.

It should be noted that the contours shown herein are only for illustrating the impact of operating and atmospheric conditions and do not represent the single-event contour of the family of aircraft described in this document. It is expected that the cumulative contours will be developed as required by planners using the data and methodology applicable to their specific study.

PRELIMINARY

7.0 PAVEMENT DATA

- 7.1 General Information**
- 7.2 Landing Gear Footprint**
- 7.3 Maximum Pavement Loads**
- 7.4 Landing Gear Loading on Pavement**
- 7.5 Flexible Pavement Requirements - U.S. Army Corps of Engineers Method S-77-1**
- 7.6 Flexible Pavement Requirements - LCN Conversion**
- 7.7 Rigid Pavement Requirements - Portland Cement Association Design Method**
- 7.8 Rigid Pavement Requirements - LCN Conversion**
- 7.9 Rigid Pavement Requirements - FAA Design Method**
- 7.10 ACN/PCN Reporting System - Flexible and Rigid Pavements**

PRELIMINARY

7.0 PAVEMENT DATA

7.1 General Information

A brief description of the pavement charts that follow will help in their use for airport planning. Each airplane configuration is depicted with a minimum range of six loads imposed on the main landing gear to aid in interpolation between the discrete values shown. All curves for any single chart represent data based on rated loads and tire pressures considered normal and acceptable by current aircraft tire manufacturer's standards. Tire pressures, where specifically designated on tables and charts, are at values obtained under loaded conditions as certificated for commercial use.

Section 7.2 presents basic data on the landing gear footprint configuration, maximum design taxi loads, and tire sizes and pressures.

Maximum pavement loads for certain critical conditions at the tire-to-ground interface are shown in Section 7.3, with the tires having equal loads on the struts.

Pavement requirements for commercial airplanes are customarily derived from the static analysis of loads imposed on the main landing gear struts. The chart in Section 7.4 is provided in order to determine these loads throughout the stability limits of the airplane at rest on the pavement. These main landing gear loads are used as the point of entry to the pavement design charts, interpolating load values where necessary.

The flexible pavement design curves (Section 7.5) are based on procedures set forth in Instruction Report No. S-77-1, "Procedures for Development of CBR Design Curves," dated June 1977, and as modified according to the methods described in ICAO Aerodrome Design Manual, Part 3, Pavements, 2nd Edition, 1983, Section 1.1 (The ACN-PCN Method), and utilizing the alpha factors approved by ICAO in October 2007. Instruction Report No. S-77-1 was prepared by the U.S. Army Corps of Engineers Waterways Experiment Station, Soils and Pavements Laboratory, Vicksburg, Mississippi. The line showing 10,000 coverages is used to calculate Aircraft Classification Number (ACN).

PRELIMINARY

The following procedure is used to develop the curves, such as shown in Section 7.5:

1. Having established the scale for pavement depth at the bottom and the scale for CBR at the top, an arbitrary line is drawn representing 10,000 coverages.
2. Values of the aircraft weights on the main landing gear are then plotted.
3. Additional annual departure lines are drawn based on the load lines of the aircraft gross weights already established.

All Load Classification Number (LCN) curves (Sections 7.6 and 7.8) have been developed from a computer program based on data provided in International Civil Aviation Organization (ICAO) document 9157-AN/901, Aerodrome Design Manual, Part 3, "Pavements," Second Edition, 1983. LCN values are shown directly for parameters of weight on main landing gear, tire pressure, and radius of relative stiffness (l) for rigid pavement or pavement thickness or depth factor (h) for flexible pavement.

Rigid pavement design curves (Section 7.7) have been prepared with the Westergaard equation in general accordance with the procedures outlined in the Design of Concrete Airport Pavement (1955 edition) by Robert G. Packard, published by the Portland Cement Association, 3800 North Wilke Road, Arlington Heights, Illinois 60004-1268. These curves are modified to the format described in the Portland Cement Association publication XP6705-2, Computer Program for Airport Pavement Design (Program PDILB), 1968, by Robert G. Packard.

PRELIMINARY

The following procedure is used to develop the rigid pavement design curves shown in Section 7.7:

1. Having established the scale for pavement thickness to the left and the scale for allowable working stress to the right, an arbitrary load line is drawn representing the main landing gear maximum weight to be shown.
2. Values of the subgrade modulus (k) are then plotted.
3. Additional load lines for the incremental values of weight on the main landing gear are drawn on the basis of the curve for $k = 300$, already established.

The rigid pavement design curves (Section 7.9) have been developed based on methods used in the FAA Advisory Circular AC 150/5320-6D, July 7, 1995. The following procedure is used to develop the curves, such as shown in Section 7.9:

1. Having established the scale for pavement flexure strength on the left and temporary scale for pavement thickness on the right, an arbitrary load line is drawn representing the main landing gear maximum weight to be shown at 5,000 coverages.
2. Values of the subgrade modulus (k) are then plotted.
3. Additional load lines for the incremental values of weight are then drawn on the basis of the subgrade modulus curves already established.
4. The permanent scale for the rigid-pavement thickness is then placed. Lines for other than 5,000 coverages are established based on the aircraft pass-to-coverage ratio.

PRELIMINARY

The ACN/PCN 40 system (Section 7.10) as referenced in ICAO Annex 14, “Aerodromes,” Fourth Edition, July, 2004, provides a standardized international airplane/pavement rating system replacing the various S, T, TT, LCN, AUW, ISWL, etc., rating systems used throughout the world. ACN is the Aircraft Classification Number and PCN is the Pavement Classification Number. An aircraft having an ACN equal to or less than the PCN can operate on the pavement subject to any limitation on the tire pressure. Numerically, the ACN is two times the derived single-wheel load expressed in thousands of kilograms, where the derived single wheel load is defined as the load on a single tire inflated to 181 psi (1.25 MPa) that would have the same pavement requirements as the aircraft. Computationally, the ACN/PCN system uses the PCA program PDILB for rigid pavements and S-77-1 for flexible pavements to calculate ACN values. The method of pavement evaluation is left up to the airport with the results of their evaluation presented as follows:

PCN	PAVEMENT TYPE	SUBGRADE CATEGORY	TIRE PRESSURE CATEGORY	EVALUATION METHOD
	R = Rigid F = Flexible	A = High B = Medium C = Low D = Ultra Low	W = No Limit X = To 217 psi (1.5 MPa) Y = To 145 psi (1.0 MPa) Z = To 73 psi (0.5 MPa)	T = Technical U = Using Aircraft

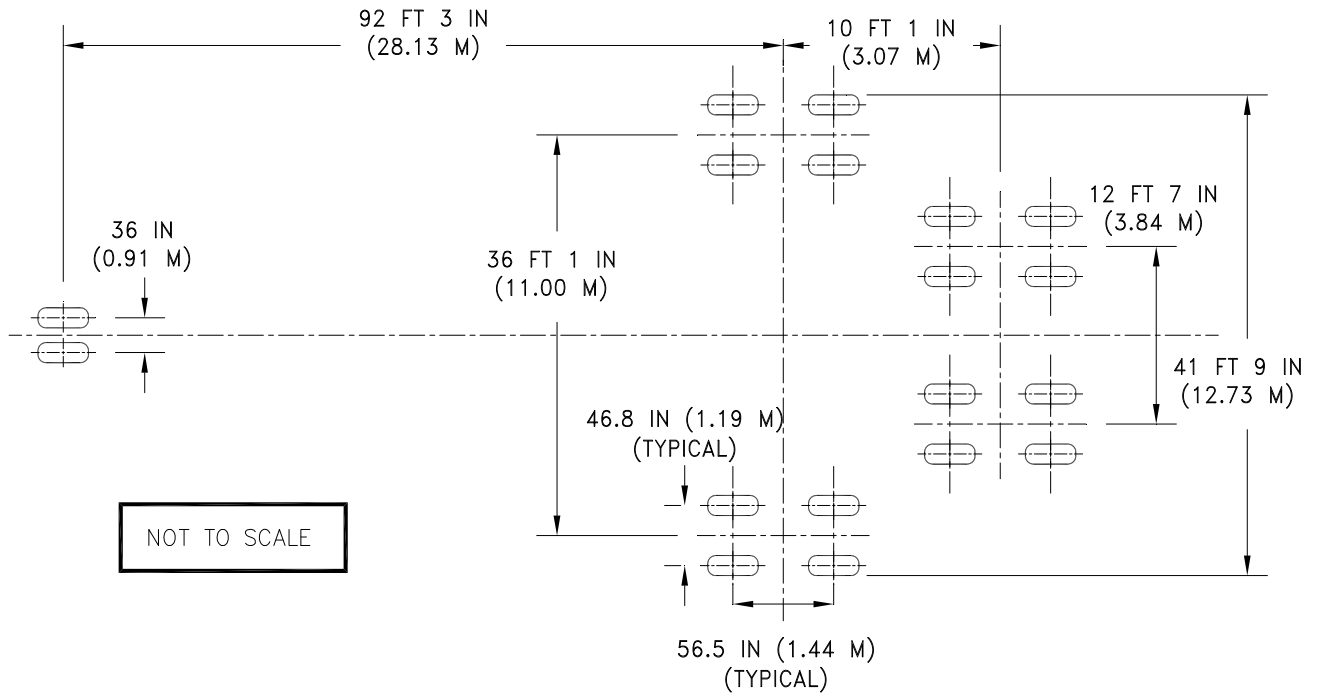
Section 7.10.1 shows the aircraft ACN values for flexible pavements. The four subgrade categories are:

- Code A - High Strength - CBR 15
- Code B - Medium Strength - CBR 10
- Code C - Low Strength - CBR 6
- Code D - Ultra Low Strength - CBR 3

Section 7.10.2 shows the aircraft ACN values for rigid pavements. The four subgrade categories are:

- Code A - High Strength, $k = 550 \text{ pci (150 MN/m}^3\text{)}$
- Code B - Medium Strength, $k = 300 \text{ pci (80 MN/m}^3\text{)}$
- Code C - Low Strength, $k = 150 \text{ pci (40 MN/m}^3\text{)}$
- Code D - Ultra Low Strength, $k = 75 \text{ pci (20 MN/m}^3\text{)}$

PRELIMINARY



	UNITS	747-8	747-8F
MAXIMUM DESIGN TAXI WEIGHT	LB KG	978,000 443,613	978,000 443,613
PERCENT OF WEIGHT ON MAIN GEAR	%	SEE SECTION 7.4	
NOSE GEAR TIRE SIZE	IN.	50 X 20.0 R 22, 26 PR	50 X 20.0 R22, 26 PR
NOSE GEAR TIRE PRESSURE	PSI KG/CM ²	166 11.67	166 11.67
MAIN GEAR TIRE SIZE	IN.	52 X 21.0 R22, 36 PR	52 X 21.0 R22, 36 PR
MAIN GEAR TIRE PRESSURE	PSI KG/CM ²	218 15.32	218 15.32

7.2 LANDING GEAR FOOTPRINT

MODEL 747-8, 747-8F

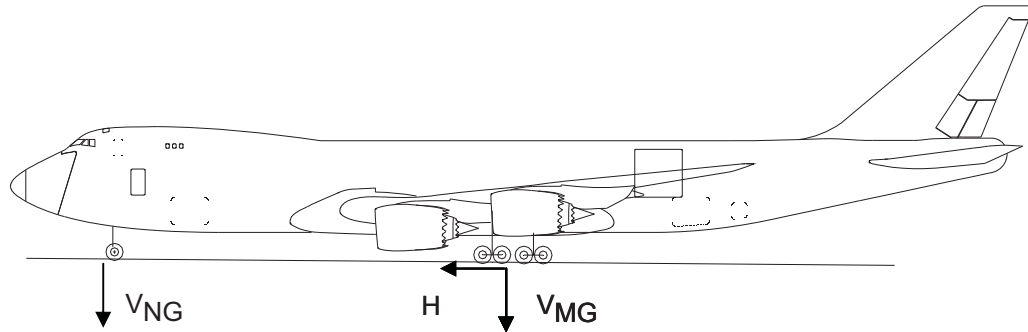
PRELIMINARY

V_{NG} = MAXIMUM VERTICAL NOSE GEAR GROUND LOAD AT MOST FORWARD CENTER OF GRAVITY

V_{MG} = MAXIMUM VERTICAL MAIN GEAR GROUND LOAD AT MOST AFT CENTER OF GRAVITY

H = MAXIMUM HORIZONTAL GROUND LOAD FROM BRAKING

NOTE: ALL LOADS CALCULATED USING AIRPLANE MAXIMUM DESIGN TAXI WEIGHT

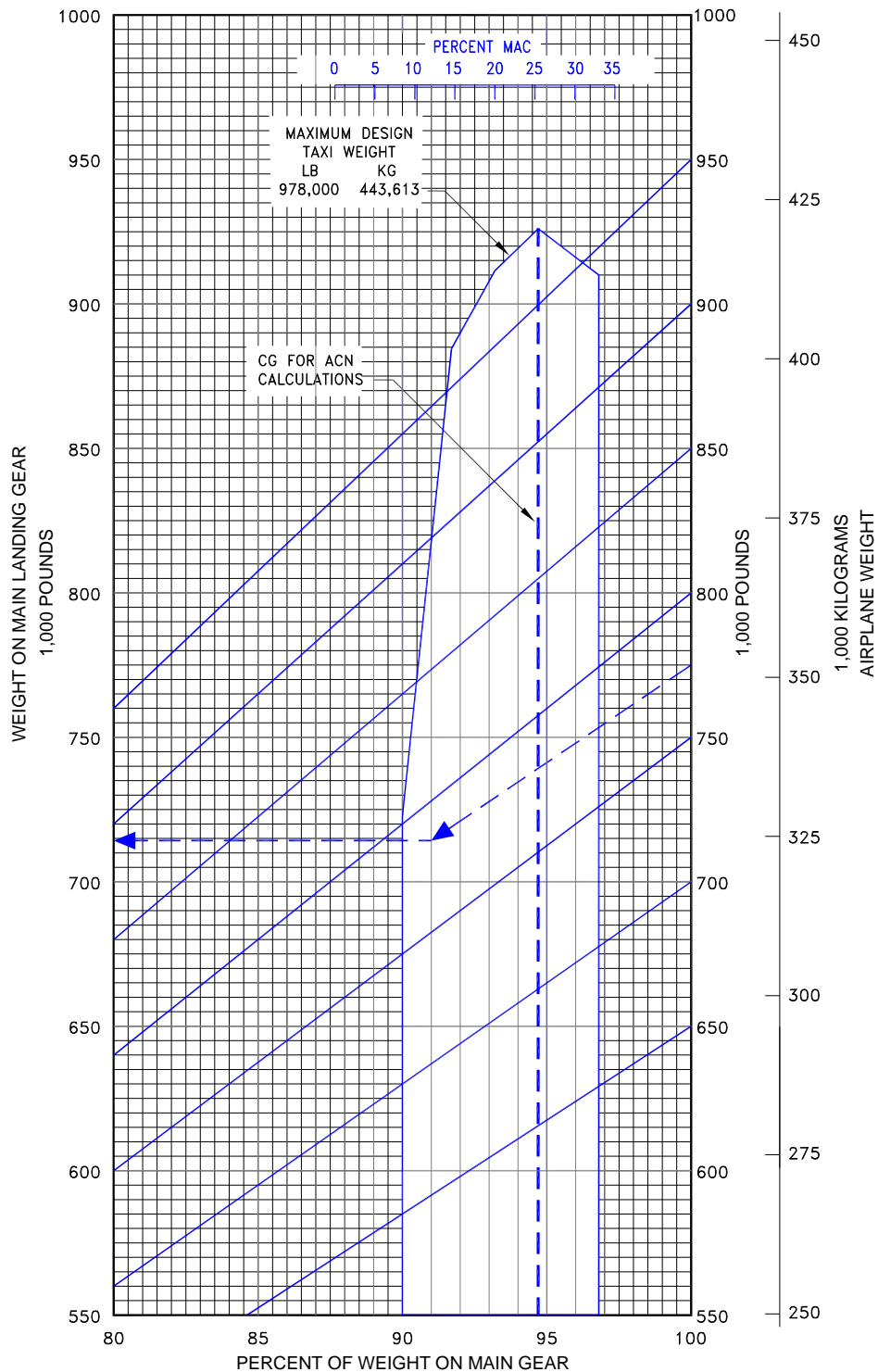


AIRPLANE MODEL	UNITS	MAX DESIGN TAXI WEIGHT	V_{NG}		V_{MG} PER STRUT (4)	H PER STRUT (4)	
			STATIC AT MOST FWD C.G.	STATIC + BRAKING 10 FT/SEC ² DECEL	MAX LOAD AT STATIC AFT C.G.	STEADY BRAKING 10 FT/SEC ² DECEL	AT INSTANTANEOUS BRAKING ($\mu = 0.8$)
747-8	LB	978,000	66,517	117,752	231,507	75,942	185,206
	KG	443,613	30,172	53,411	105,010	34,447	84,008
747-8F	LB	978,000	65,145	116,380	231,507	75,942	185,206
	KG	443,613	29,549	52,789	105,010	34,447	84,008

7.3. MAXIMUM PAVEMENT LOADS

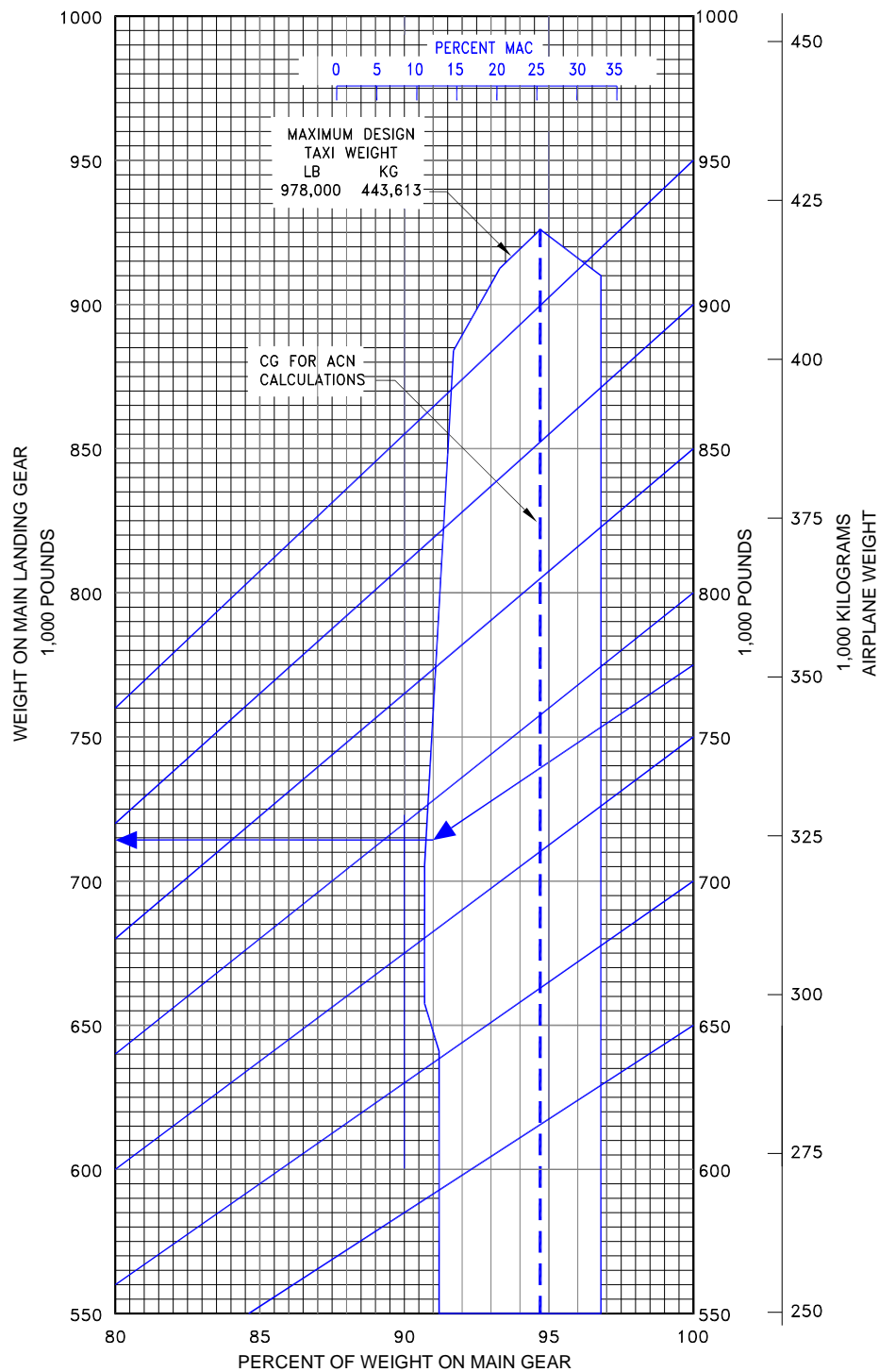
MODEL 747-8, -747-8F

PRELIMINARY



7.4.1 LANDING GEAR LOADING ON PAVEMENT MODEL 747-8

PRELIMINARY



7.4.2 LANDING GEAR LOADING ON PAVEMENT MODEL 747-8F

PRELIMINARY

7.5 Flexible Pavement Requirements - U.S. Army Corps of Engineers Method (S-77-1)

The following flexible-pavement design chart presents the data of six incremental main-gear loads at the minimum tire pressure required at the maximum design taxi weight.

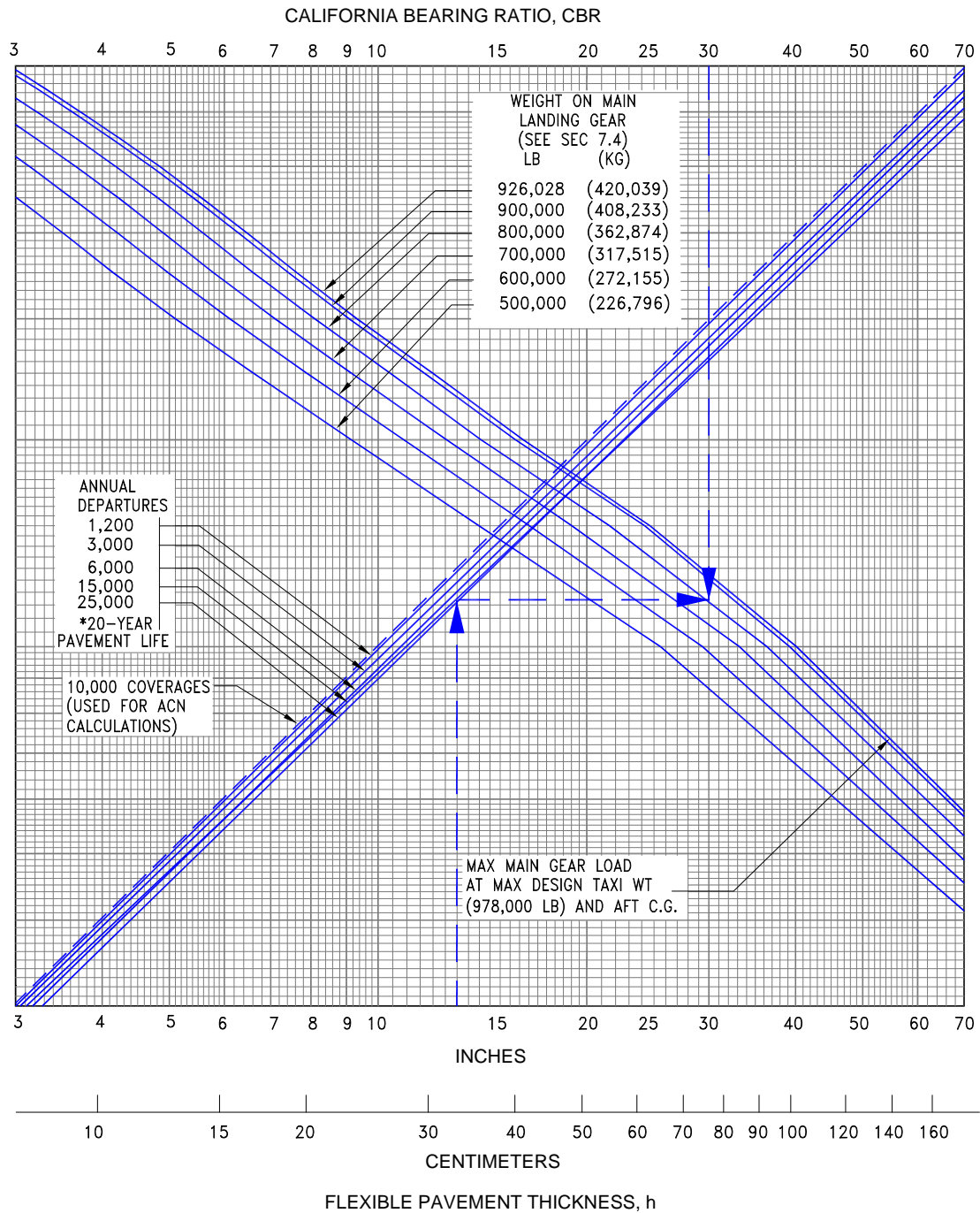
In the example shown in Section 7.5, for a CBR of 30 and an annual departure level of 15,000, the required flexible pavement thickness for an airplane with a main gear loading of 800,000 pounds is 13.0 inches.

The line showing 10,000 coverages is used for ACN calculations (see Section 7.10).

The FAA design method uses a similar procedure using total airplane weight instead of weight on the main landing gears. The equivalent main gear loads for a given airplane weight could be calculated from Section 7.4.

PRELIMINARY

Note: TIRES – 52 x 21 R22, 36PR AT 218 PSI (15.32 KG/CM SQ)



7.5.1 FLEXIBLE PAVEMENT REQUIREMENTS - U.S. ARMY CORPS OF ENGINEERS DESIGN METHOD (S-77-1) MODEL 747-8, 747-8F

PRELIMINARY

7.6 Flexible Pavement Requirements - LCN Method

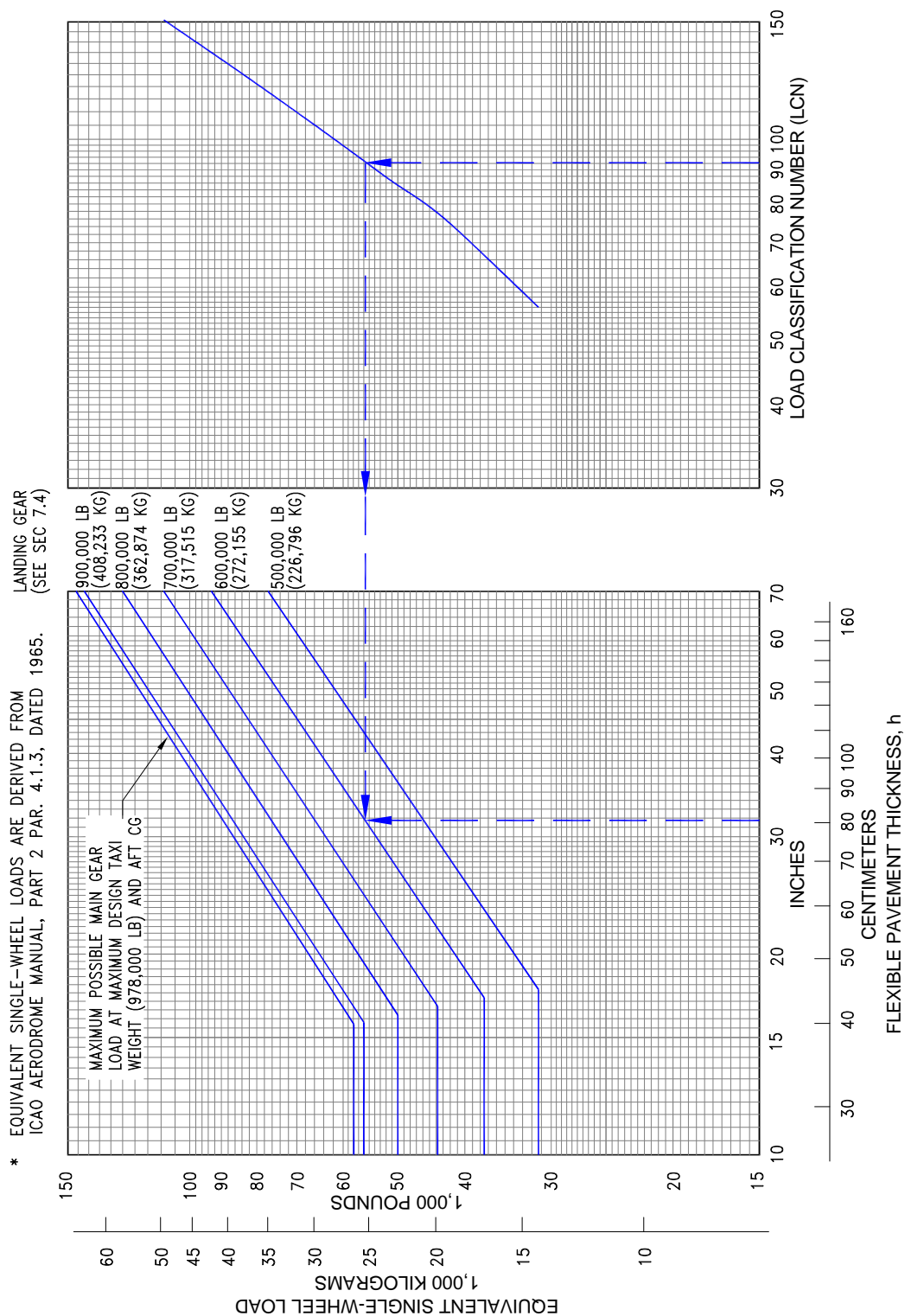
To determine the airplane weight that can be accommodated on a particular flexible pavement, both the Load Classification Number (LCN) of the pavement and the thickness must be known.

In the example shown in Section 7.6, flexible pavement thickness is shown at 32 in. with an LCN of 92. For these conditions, the apparent maximum allowable weight permissible on the main landing gear is 600,000 lb for an airplane with 218-psi main gear tires.

Note: If the resultant aircraft LCN is not more than 10% above the published pavement LCN, the bearing strength of the pavement can be considered sufficient for unlimited use by the airplane. The figure 10% has been chosen as representing the lowest degree of variation in LCN that is significant (reference: ICAO Aerodrome Design Manual, Part 2, "Aerodrome Physical Characteristics," Chapter 4, Paragraph 4.1.5.7v, 2nd Edition dated 1965).

PRELIMINARY

Note: TIRES – 52 x 21 R22, 36PR AT 218 PSI (15.32 KG/CM SQ)



7.6.1 FLEXIBLE PAVEMENT REQUIREMENTS - LCN METHOD

MODEL 747-8, 747-8F

PRELIMINARY

7.7 Rigid Pavement Requirements - Portland Cement Association Design Method

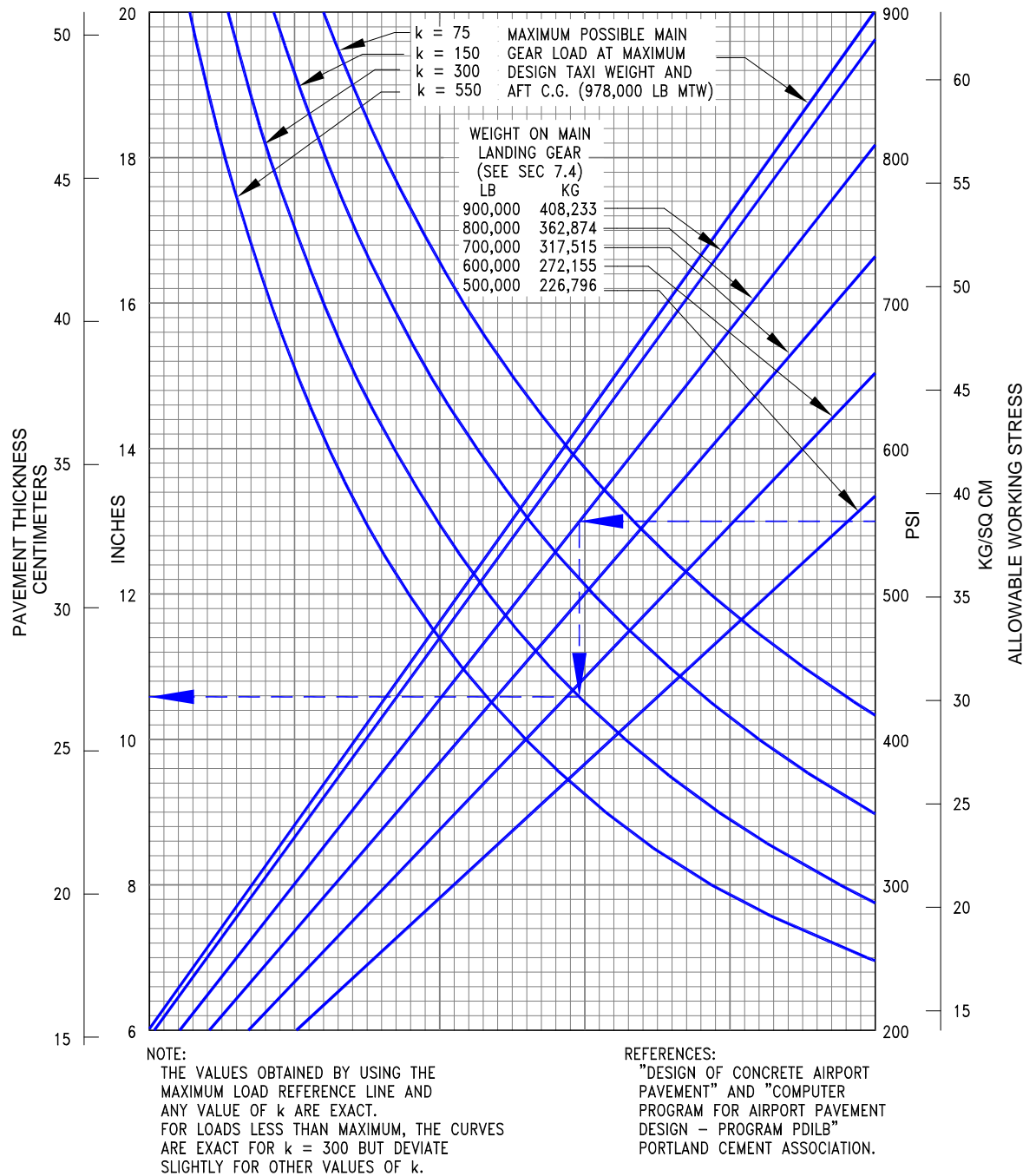
The Portland Cement Association method of calculating rigid pavement requirements is based on the computerized version of "Design of Concrete Airport Pavement" (Portland Cement Association, 1965) as described in XP6705-2, "Computer Program for Airport Pavement Design" by Robert G. Packard, Portland Cement Association, 1968.

The rigid pavement design chart in Section 7.7 presents the data for six incremental main gear loads at the minimum tire pressure required at the maximum design taxi weight.

In the example shown, for an allowable working stress of 550 psi, a main gear load of 800,000 lb, and a subgrade strength (k) of 300, the required rigid pavement thickness is 10.6 in.

PRELIMINARY

Note: TIRES – 52 x 21 R22, 36PR AT 218 PSI (15.32 KG/CM SQ)



7.7.1 RIGID PAVEMENT REQUIREMENTS - PORTLAND CEMENT ASSOCIATION DESIGN METHOD

MODEL 747-8, 747-8F

PRELIMINARY

7.8 Rigid Pavement Requirements - LCN Conversion

To determine the airplane weight that can be accommodated on a particular rigid pavement, both the LCN of the pavement and the radius of relative stiffness (l) of the pavement must be known.

In the example shown in Section 7.8.2, for a rigid pavement with a radius of relative stiffness of 47 with an LCN of 91, the apparent maximum allowable weight permissible on the main landing gear is 600,000 lb for an airplane with 218-psi main tires.

Note: If the resultant aircraft LCN is not more than 10% above the published pavement LCN, the bearing strength of the pavement can be considered sufficient for unlimited use by the airplane. The figure 10% has been chosen as representing the lowest degree of variation in LCN that is significant (reference: ICAO Aerodrome Design Manual, Part 2, "Aerodrome Physical Characteristics," Chapter 4, Paragraph 4.1.5.7v, 2nd Edition dated 1965).

PRELIMINARY

RADIUS OF RELATIVE STIFFNESS (ℓ)

VALUES IN INCHES

$$\ell = \sqrt[4]{\frac{Ed^3}{12(1-\mu^2)k}} = 24.1652 \sqrt[4]{\frac{d^3}{k}}$$

WHERE: E = YOUNG'S MODULUS OF ELASTICITY = 4×10^6 psi

k = SUBGRADE MODULUS, LB PER CU IN

d = RIGID PAVEMENT THICKNESS, IN

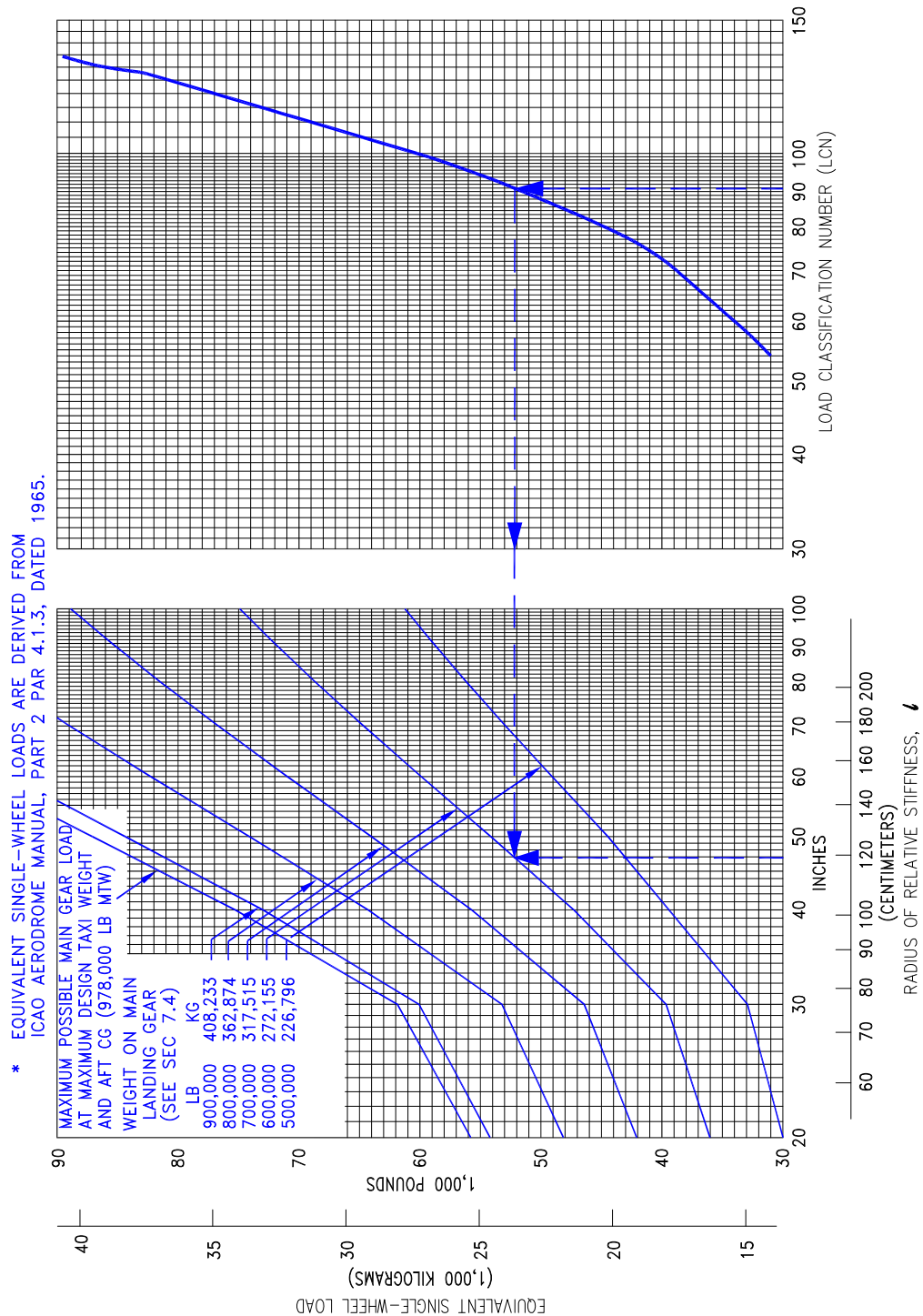
μ = POISSON'S RATIO = 0.15

d	k = 75	k = 100	k = 150	k = 200	k = 250	k = 300	k = 350	k = 400	k = 500	k = 550
6.0	31.48	29.29	26.47	24.63	23.30	22.26	21.42	20.71	19.59	19.13
6.5	33.42	31.10	28.11	26.16	24.74	23.63	22.74	21.99	20.80	20.31
7.0	35.33	32.88	29.71	27.65	26.15	24.99	24.04	23.25	21.99	21.47
7.5	37.21	34.63	31.29	29.12	27.54	26.31	25.32	24.49	23.16	22.61
8.0	39.06	36.35	32.84	30.56	28.91	27.62	26.57	25.70	24.31	23.73
8.5	40.87	38.04	34.37	31.99	30.25	28.90	27.81	26.90	25.44	24.84
9.0	42.66	39.70	35.88	33.39	31.57	30.17	29.03	28.07	26.55	25.93
9.5	44.43	41.35	37.36	34.77	32.88	31.42	30.23	29.24	27.65	27.00
10.0	46.17	42.97	38.83	36.13	34.17	32.65	31.41	30.38	28.73	28.06
10.5	47.89	44.57	40.27	37.48	35.44	33.87	32.58	31.52	29.81	29.10
11.0	49.59	46.15	41.70	38.81	36.70	35.07	33.74	32.63	30.86	30.14
11.5	51.27	47.72	43.12	40.12	37.95	36.26	34.89	33.74	31.91	31.16
12.0	52.94	49.26	44.51	41.43	39.18	37.43	36.02	34.83	32.94	32.17
12.5	54.58	50.80	45.90	42.71	40.40	38.60	37.14	35.92	33.97	33.17
13.0	56.21	52.31	47.27	43.99	41.60	39.75	38.25	36.99	34.98	34.16
13.5	57.83	53.81	48.63	45.25	42.80	40.89	39.34	38.05	35.99	35.14
14.0	59.43	55.30	49.97	46.50	43.98	42.02	40.43	39.10	36.98	36.11
14.5	61.01	56.78	51.30	47.74	45.15	43.14	41.51	40.15	37.97	37.07
15.0	62.58	58.24	52.62	48.97	46.32	44.25	42.58	41.18	38.95	38.03
15.5	64.14	59.69	53.93	50.19	47.47	45.35	43.64	42.21	39.92	38.98
16.0	65.69	61.13	55.23	51.40	48.61	46.45	44.69	43.22	40.88	39.92
16.5	67.22	62.55	56.52	52.60	49.75	47.53	45.73	44.23	41.83	40.85
17.0	68.74	63.97	57.80	53.79	50.87	48.61	46.77	45.23	42.78	41.77
17.5	70.25	65.38	59.07	54.97	51.99	49.68	47.80	46.23	43.72	42.69
18.0	71.75	66.77	60.34	56.15	53.10	50.74	48.82	47.22	44.65	43.60
19.0	74.72	69.54	62.83	58.47	55.30	52.84	50.84	49.17	46.50	45.41
20.0	77.65	72.26	65.30	60.77	57.47	54.91	52.83	51.10	48.33	47.19
21.0	80.55	74.96	67.73	63.03	59.61	56.95	54.80	53.00	50.13	48.95
22.0	83.41	77.62	70.14	65.27	61.73	58.98	56.75	54.88	51.91	50.68
23.0	86.23	80.25	72.51	67.48	63.82	60.98	58.67	56.74	53.67	52.40
24.0	89.03	82.85	74.86	69.67	65.89	62.95	60.57	58.58	55.41	54.10
25.0	91.80	85.43	77.19	71.84	67.94	64.91	62.46	60.41	57.13	55.78

7.8.1 RADIUS OF RELATIVE STIFFNESS (REFERENCE: PORTLAND CEMENT ASSOCIATION)

PRELIMINARY

Note: TIRES – 52 x 21 R22, 36PR AT 218 PSI (15.32 KG/CM SQ)



7.8.2 RIGID PAVEMENT REQUIREMENTS - LCN CONVERSION

MODEL 747-8, 747-8F

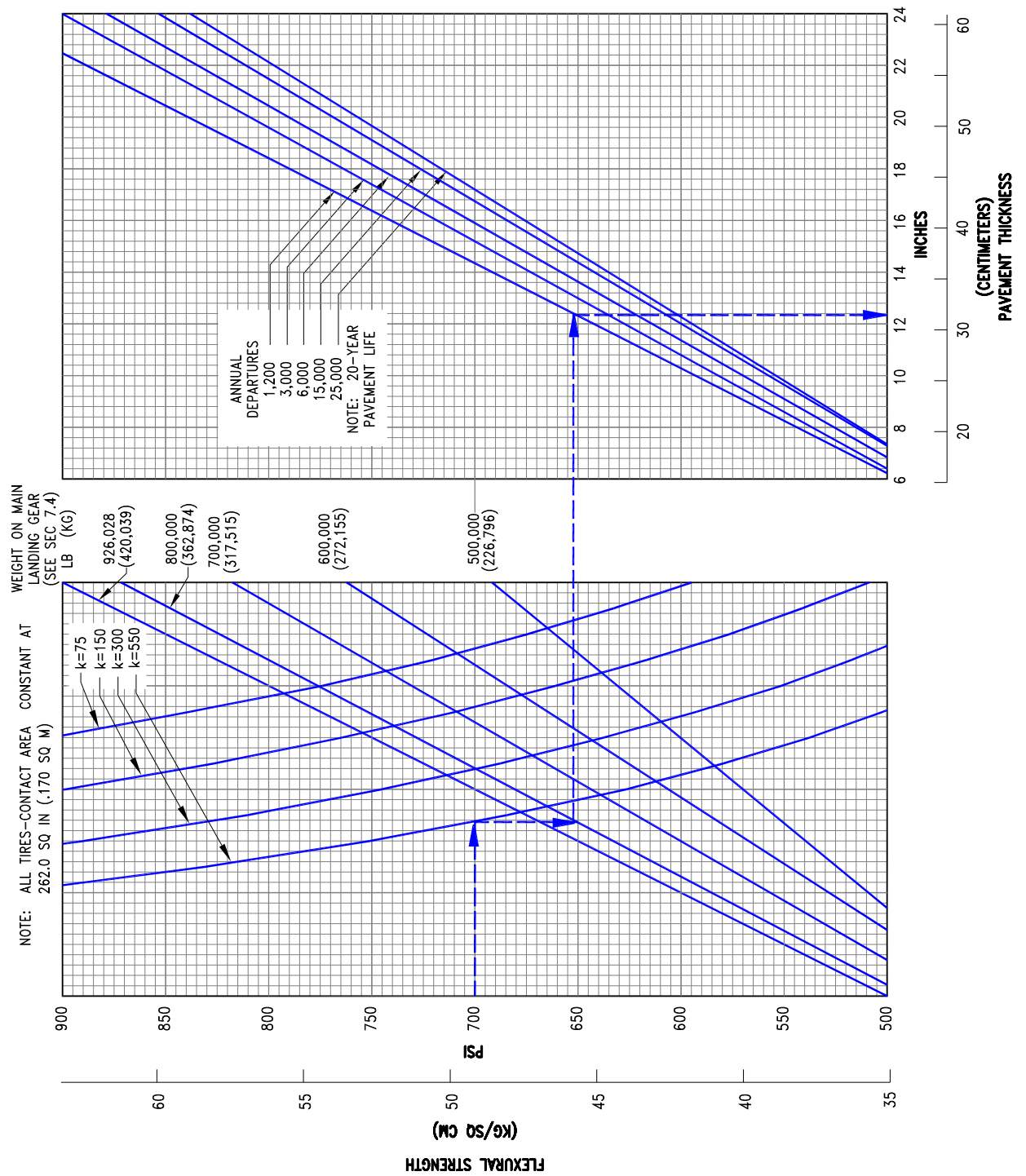
PRELIMINARY

7.9 Rigid Pavement Requirements - FAA Design Method

The rigid pavement design chart shown in Section 7.9 presents the data of five incremental main gear loads at the minimum tire pressure required at the maximum design taxi weight.

In the example shown, for a pavement flexure strength of 700 psi, a subgrade strength of $k = 550$, and an annual departure level of 1,200, the required rigid pavement thickness for an airplane with a main gear load of 800,000 lb is 12.4 in.

PRELIMINARY



7.9.1 RIGID PAVEMENT REQUIREMENTS - FAA DESIGN METHOD MODEL 747-8, 747-8F

PRELIMINARY

7.10 ACN/PCN Reporting System: Flexible and Rigid Pavements

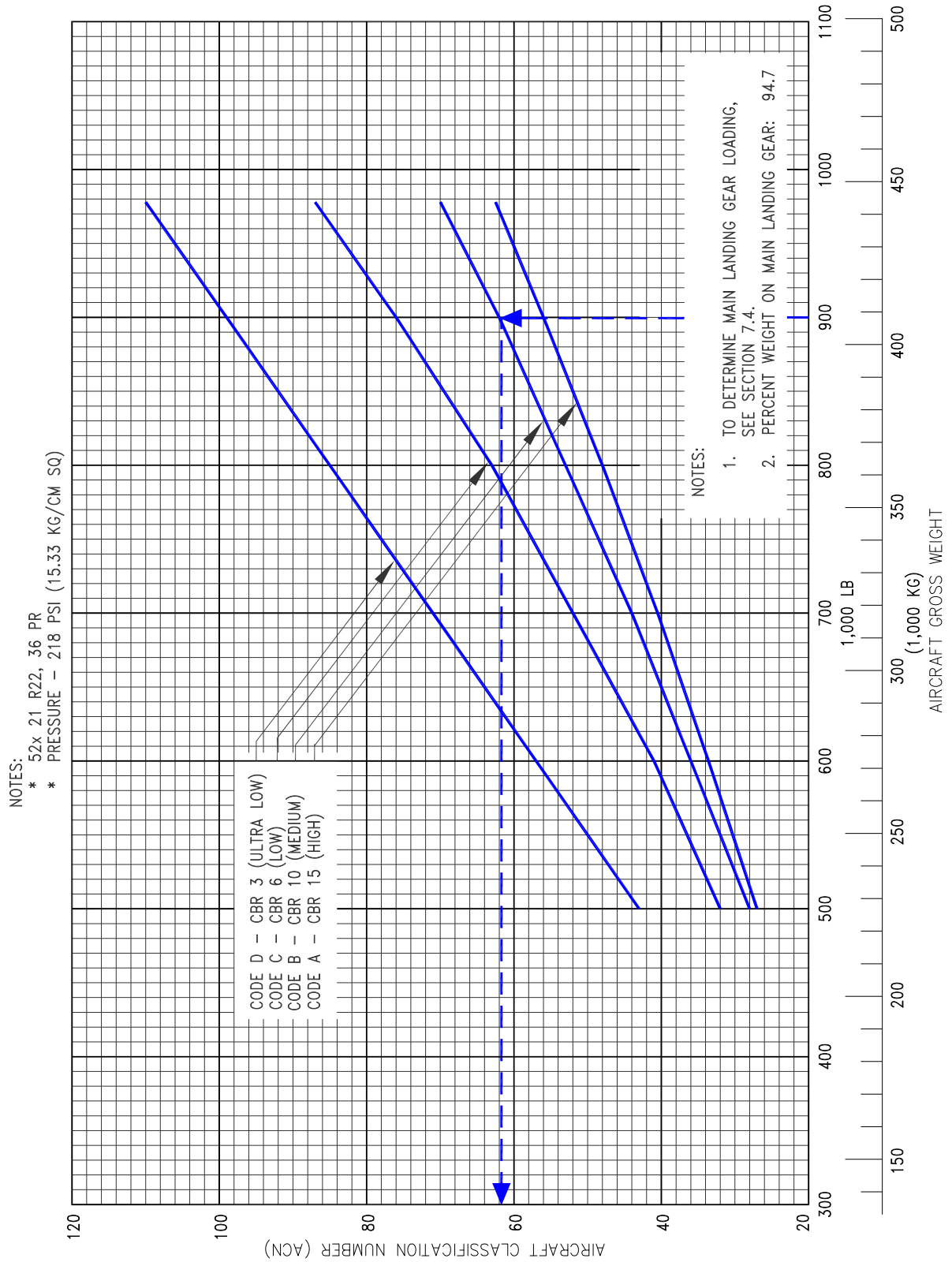
To determine the ACN of an aircraft on flexible or rigid pavement, both the aircraft gross weight and the subgrade strength category must be known. In the chart in Section 7.10.1, for an aircraft with gross weight of 900,000 lb and medium subgrade strength, the flexible pavement ACN is 62. In Section 7.10.2, for the same gross weight and subgrade strength, the rigid pavement ACN is 67.

The following table provides ACN data in tabular format similar to the one used by ICAO in the “Aerodrome Design Manual Part 3, Pavements”. If the ACN for an intermediate weight between maximum taxi weight and the empty weight of the aircraft is required, Figures 7.10.1 through 7.10.2 should be consulted.

AIRCRAFT TYPE	MAXIMUM TAXI WEIGHT MINIMUM WEIGHT (1) LB (KG)	LOAD ON ONE MAIN GEAR LEG (%)	TIRE PRESSURE PSI (MPa)	ACN FOR RIGID PAVEMENT SUBGRADES – MN/m ³				ACN FOR FLEXIBLE PAVEMENT SUBGRADES – CBR			
				HIGH 150	MEDIUM 80	LOW 40	ULTRA LOW 20	HIGH 15	MEDIUM 10	LOW 6	ULTRA LOW 3
747-8, -8F	978,000(443,613)	23.68	218 (1.50)	64	75	88	101	63	70	87	110
	500,000(226,796)			27	30	35	41	27	28	32	43

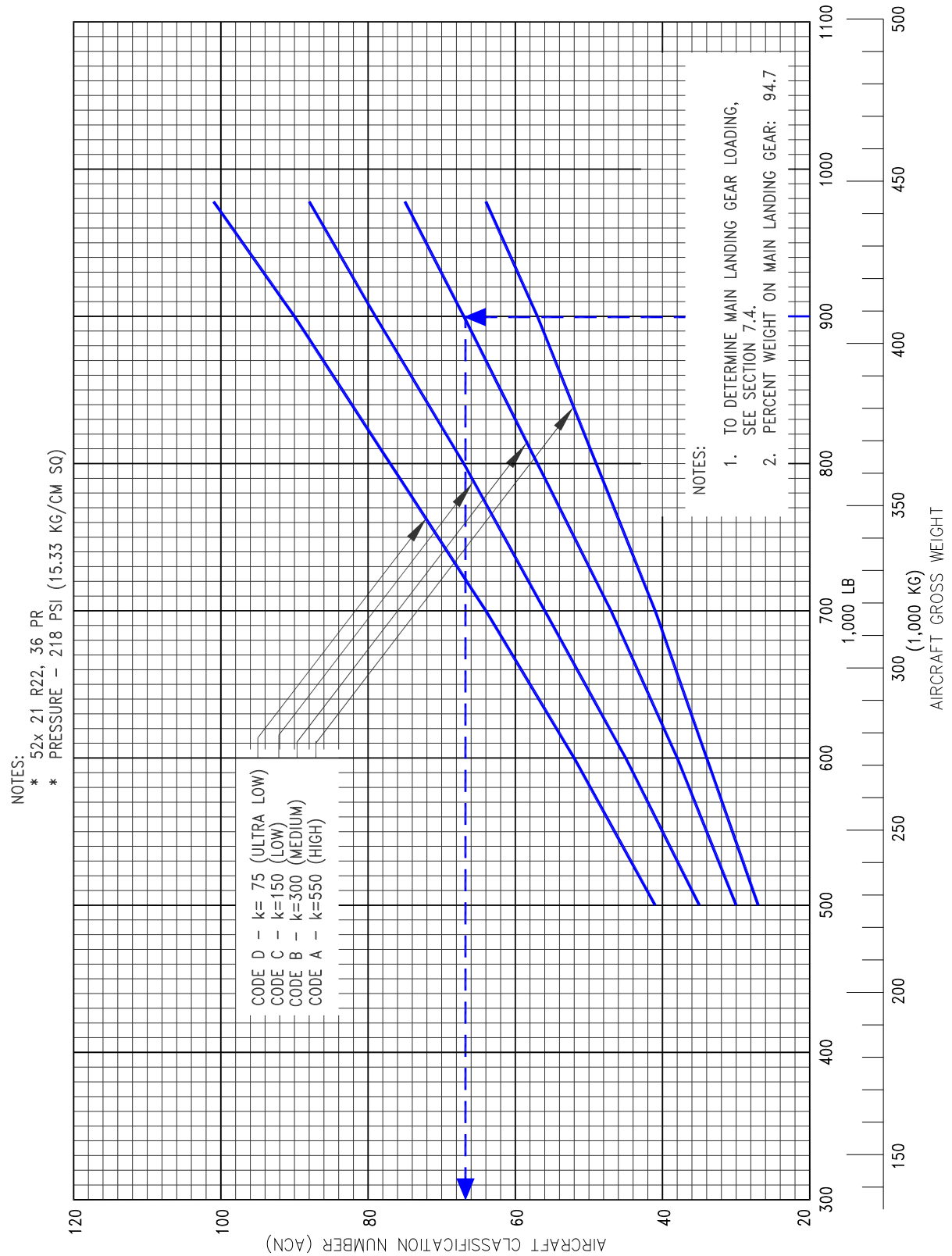
(1) Minimum weight used solely as a baseline for ACN curve generation.

PRELIMINARY



7.10.1 AIRCRAFT CLASSIFICATION NUMBER - FLEXIBLE PAVEMENT MODEL 747-8, 747-8F

PRELIMINARY



7.10.2 AIRCRAFT CLASSIFICATION NUMBER - RIGID PAVEMENT MODEL 747-8, 747-8F

PRELIMINARY

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PRELIMINARY

8.0 FUTURE 747-8 DERIVATIVE AIRPLANES

PRELIMINARY

8.0 FUTURE 747-8 DERIVATIVE AIRPLANES

Several derivatives are being studied to provide additional capabilities of the 747-8 family of airplanes. Future growth versions could address additional passenger count, cargo capacity, increased range, or environmental performance.

Whether and/or when these or other possibilities are actually built is entirely dependent on future airline requirements. In any event, the impact on airport facilities will be a consideration in configuration and design.

PRELIMINARY

9.0 SCALED 747-8 DRAWINGS

9.1 747-8, 747-8F

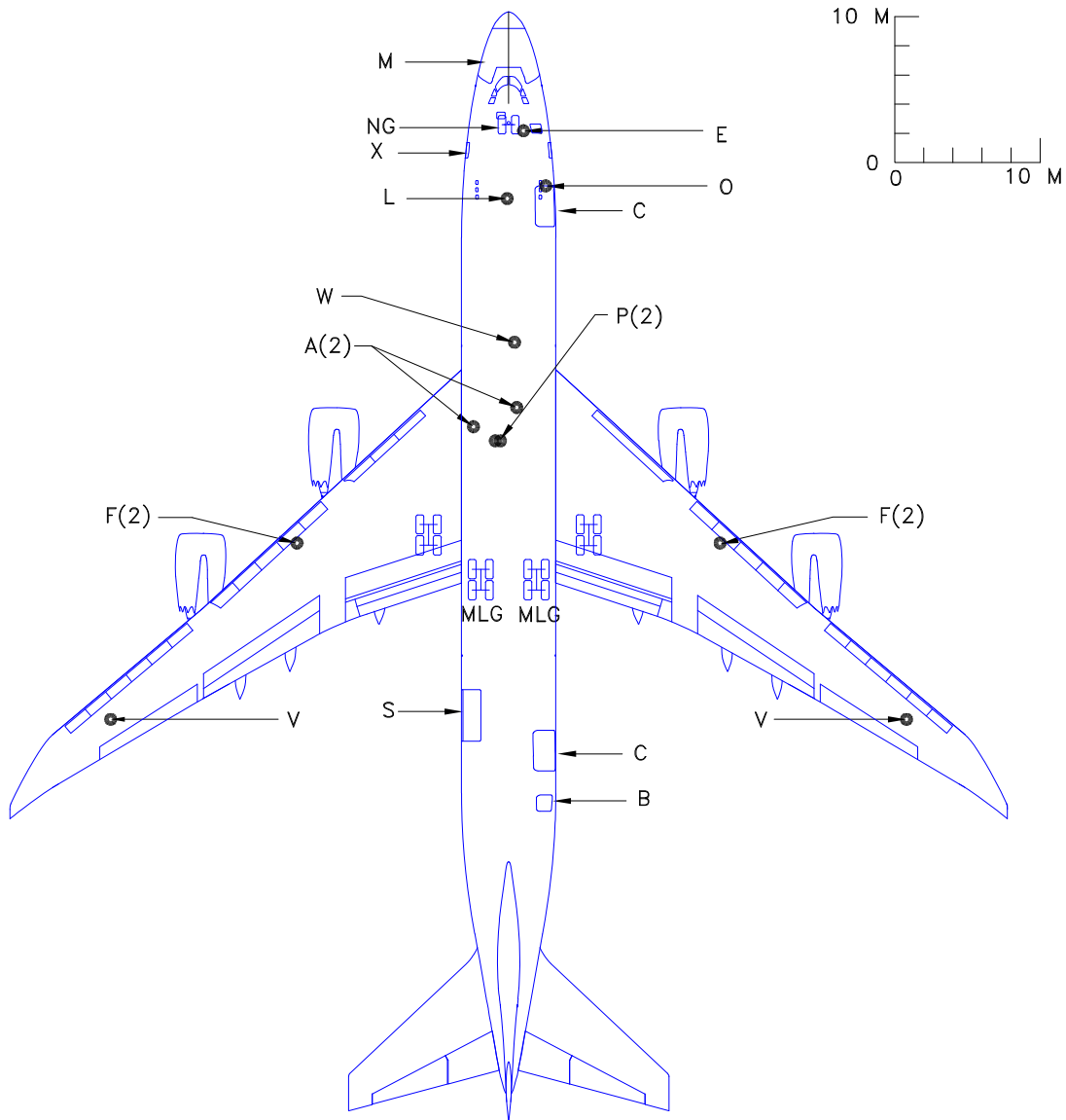
PRELIMINARY

9.0 SCALED DRAWINGS

The drawings in the following pages show airplane plan view drawings, drawn to approximate scale as noted. The drawings may not come out to exact scale when printed or copied from this document. Printing scale should be adjusted when attempting to reproduce these drawings. Three-view drawing files of the 747-8 Freighter, along with other Boeing airplane models, may be downloaded from the following website:

<http://www.boeing.com/airports>

PRELIMINARY



LEGEND

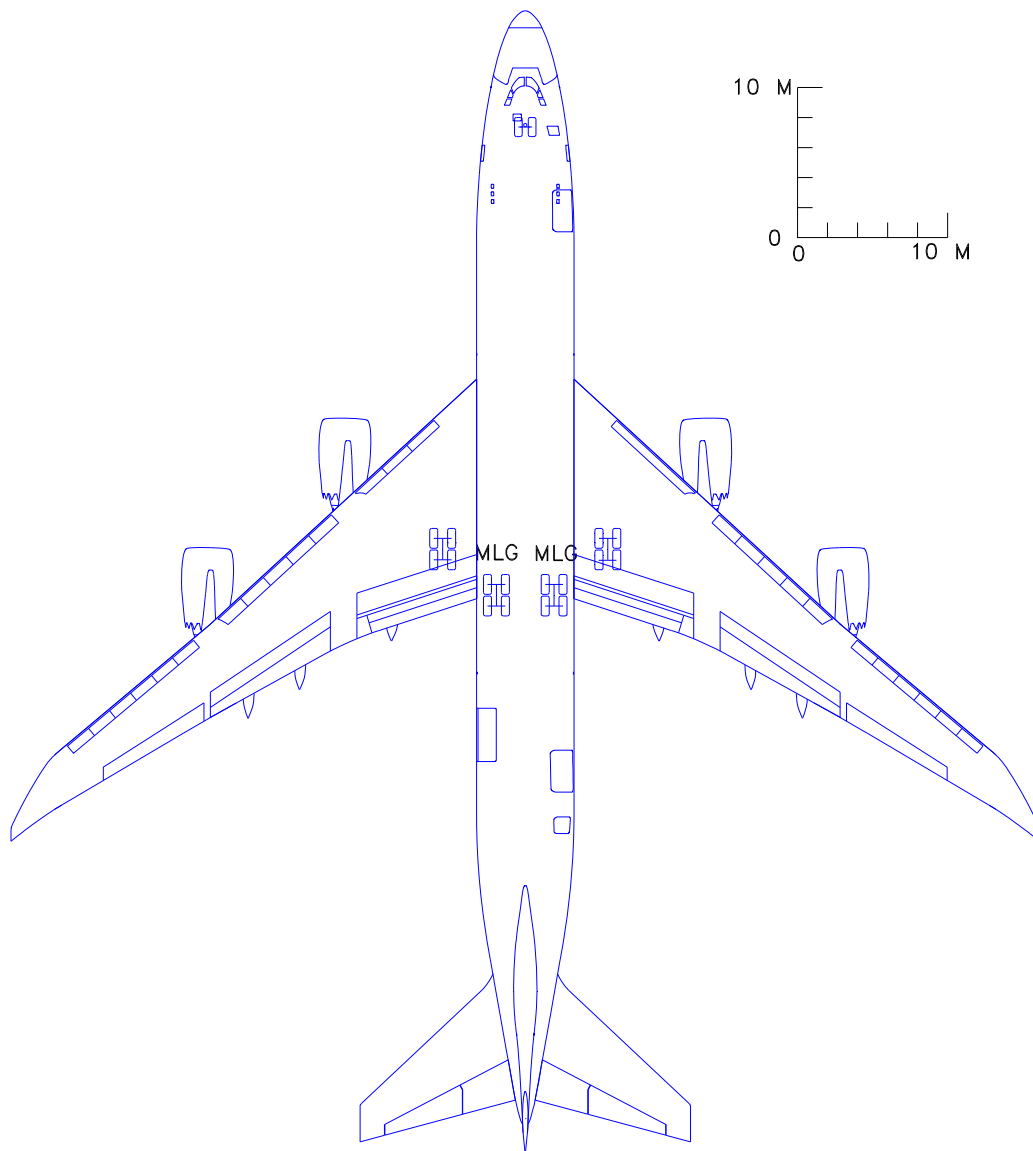
A(2)	AIR CONDITIONING (2 CONNECTIONS)	NG	NOSE GEAR
B	BULK CARGO DOOR	O	OXYGEN
C	CARGO CONTAINER DOOR	P(2)	PNEUMATIC (2 CONNECTIONS)
E(2)	ELECTRICAL (2 CONNECTIONS)	S	MAIN DECK SIDE CARGO DOOR
F(2)	FUEL (2 CONNECTIONS)	U	UPPER DECK EXIT DOOR
L	LAVATORY	V	FUEL VENT
M	MAIN DECK NOSE CARGO DOOR	W	POTABLE WATER
MLG	MAIN LANDING GEAR	X	PASSENGER DOOR

NOTE: ADJUST FOR PROPER SCALING WHEN PRINTING THIS PAGE

9.1.1 SCALED DRAWING - 1:500

MODEL 747-8, 747-8F

PRELIMINARY



NOTE: ADJUST FOR PROPER SCALING WHEN PRINTING THIS PAGE

9.1.2 SCALED DRAWING - 1:500 *MODEL 747-8, 747-8F*