

GRAVITATION

★★

A Satellite moving in a Circular orbit of radius ($r = nR$) around Earth. Resistive force acting on the

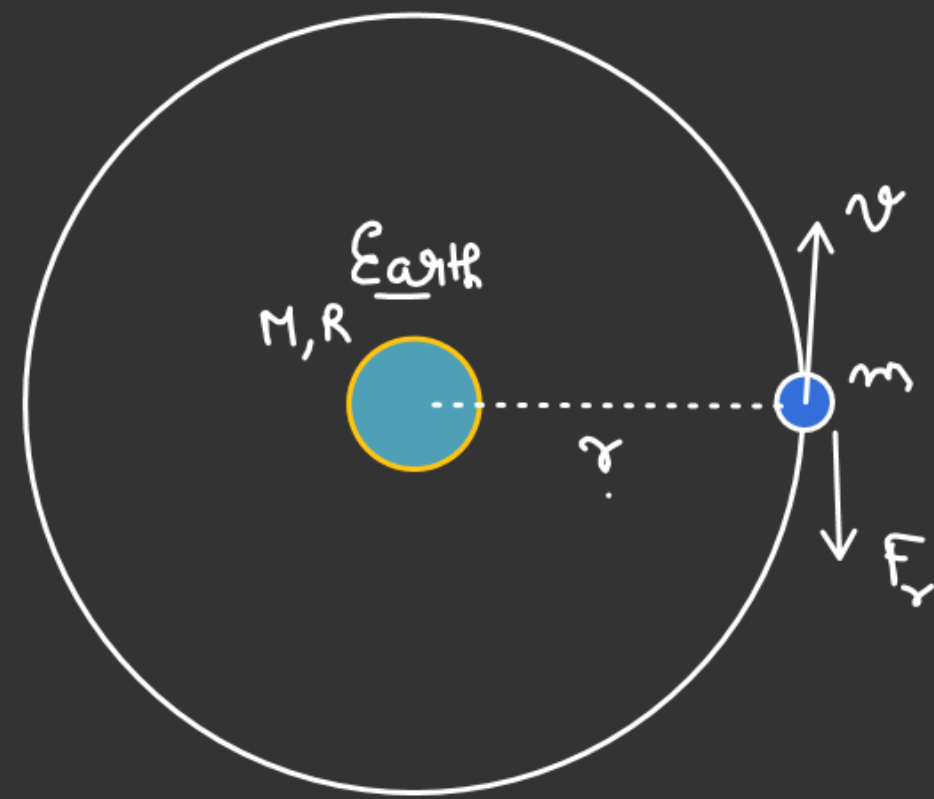
Satellite $F_r = \alpha v^2$

How long it takes for Satellite to collide with earth.

Total Energy of Satellite at an orbital radius r is $E_T = \left(-\frac{GMm}{2r} \right)$

$$\left(\text{Rate of loss of Energy of Satellite} \right) = -(F \cdot v)$$

$$-\left(\frac{dE_T}{dt} \right) = -fv$$



GRAVITATION

$$\left(\frac{dE_T}{dt}\right) = -\underline{f v} \quad \checkmark$$

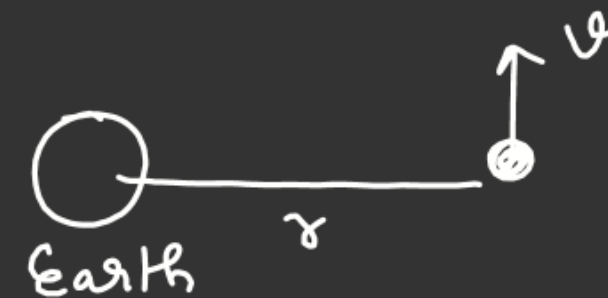
$$\frac{d}{dt} \left(-\frac{GMm}{2r} \right) = -(\alpha v^2) v$$

$$+ \frac{GMm}{2} \left(-\frac{1}{r^2} \right) \left(\frac{dr}{dt} \right) = \alpha v^3$$

$$-\frac{GMm}{2r^2} \left(\frac{dr}{dt} \right) = \alpha \left(\frac{v^2}{r} \cdot \sqrt{\frac{GM}{r}} \right)$$

$$- \frac{1}{2} \sqrt{\frac{1}{GM}} \int_{rR}^R r^{-1/2} dr = \frac{3}{2} \int_0^t dt$$

$$t = \frac{3m}{2} \sqrt{\frac{R}{GM}} (\sqrt{n} - 1) \quad \underline{\text{Ans}} \quad \checkmark$$



$$v_r = \sqrt{\frac{GM}{r}}$$

orbital
velocity of Satellite
at any radial distance
 r

GRAVITATION

Assume Circular orbit of Earth around Sun. Assume Earth Suddenly Stop rotating. Find time when earth collapse in Sun. $T = (\text{Time period of Earth about Sun})$

$$-\frac{GMm}{r} = -\frac{GMm}{x} + \frac{1}{2}mv^2$$

$$\frac{1}{2}mv^2 = GMm \left(\frac{1}{x} - \frac{1}{r} \right)$$

$$v^2 = \frac{2GM}{r} \left(\frac{r-x}{x} \right)$$

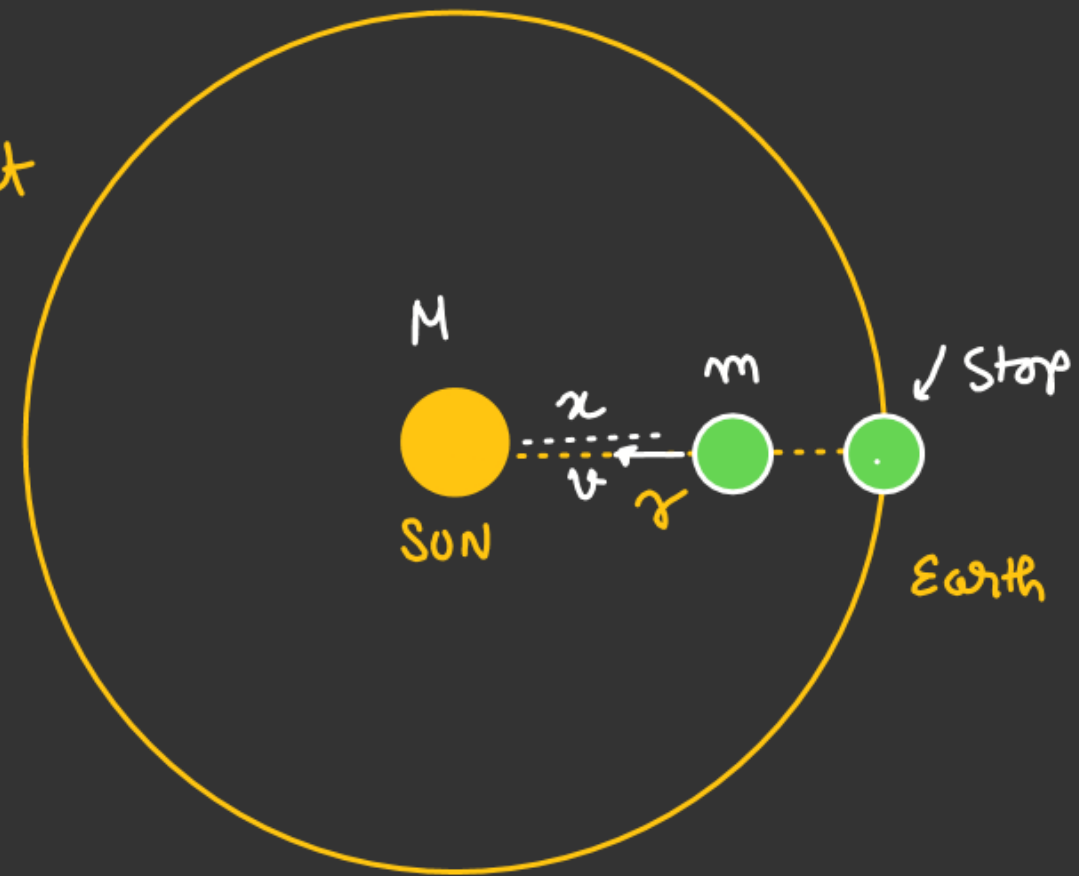
$$v = \sqrt{\frac{2GM}{r}} \sqrt{\frac{r-x}{x}}$$

$$-\frac{dx}{dt} = \sqrt{\frac{2GM}{r}} \sqrt{\frac{r-x}{x}}$$

$$\int_r^0 \sqrt{\frac{x}{r-x}} dx = -\sqrt{\frac{2GM}{r}} \int_0^t dt$$

put $x = r \sin^2 \theta$

$$(t = \frac{T}{2\sqrt{8}})$$



GRAVITATION

Assume Circular orbit of Earth around Sun. Assume Earth Suddenly Stop rotating. Find time when earth collapse in Sun. $T = (\text{Time period of Earth about Sun})$

Assume motion of Earth as elliptical
let, T' be its time period and
 $\frac{r}{2}$ be the semi-major axis.

By Kepler's 3rd law.

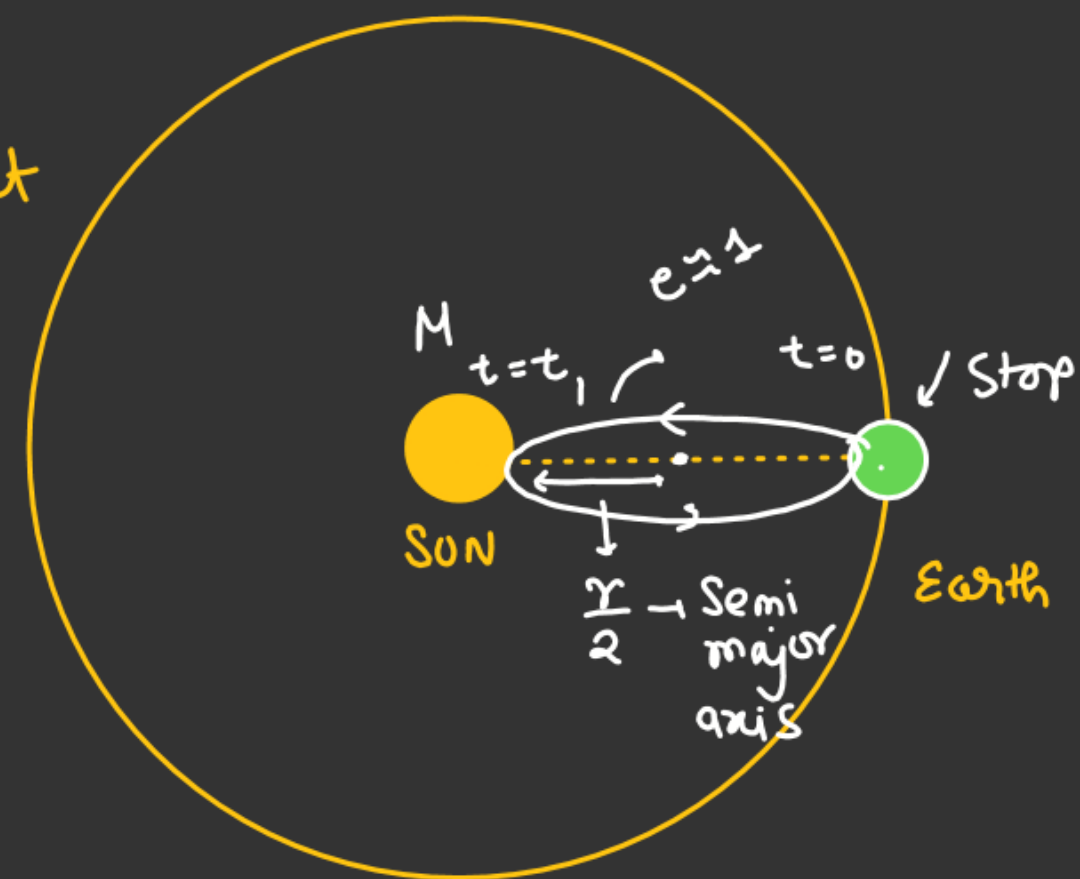
$$\frac{(T')^2}{(T)^2} = \frac{(r/2)^3}{(r)^3}$$

$$\frac{(T')^2}{T^2} = \frac{1}{8} \Rightarrow T' = \frac{T}{\sqrt{8}}$$

$$t_1 = \frac{T'}{2}$$

$$t_1 = \frac{T}{2\sqrt{8}}$$

Ans.



$$\frac{b \rightarrow 0 \quad e \rightarrow 1}{\text{Diagram of a very flat ellipse}} \rightarrow$$

GRAVITATION

AA: A Body is projected at an angle θ from horizontal. from the surface of earth with velocity v_0 . Show that the semi-major axis of ellipse is independent of θ .

Let, r be either min or maximum distance of the body.

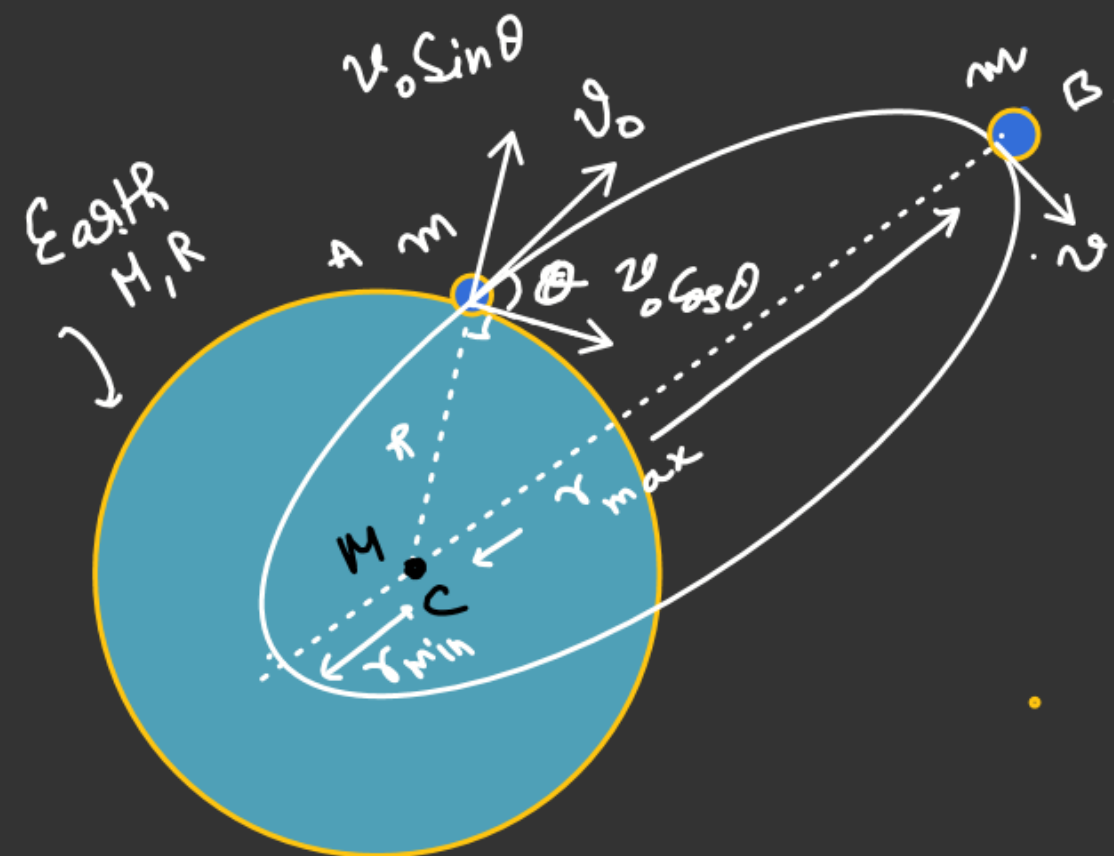
A.M.C about C from A to B.

$$(mv_0 \cos \theta) R = mvr \quad \text{--- (1)}$$

Energy Conservation from A to B.

$$-\frac{GMm}{R} + \frac{1}{2}mv_0^2 = -\frac{GMm}{r} + \frac{1}{2}mv^2 \quad \text{--- (2)}$$

From (1) $v = (Rv_0 \cos \theta)$



Semimajor axis = $\left(\frac{r_{\max} + r_{\min}}{2} \right)$

GRAVITATION

$$(mv_0 \cos \theta) R = mvr \quad \text{--- (1)}$$

Energy Conservation from A to B.

$$-\frac{GMm}{R} + \frac{1}{2}mv_0^2 = -\frac{GMm}{r} + \frac{1}{2}mv^2 \quad \text{--- (2)}$$

From (1) $v = \left(\frac{Rv_0 \cos \theta}{r} \right)$

$$-\frac{GMm}{R} + \frac{mv_0^2}{2} = -\frac{GMm}{r} + \frac{m}{2} \left(\frac{R^2 v_0^2 \cos^2 \theta}{r^2} \right)$$

$$GMm \left(\frac{1}{r} - \frac{1}{R} \right) = \frac{mv_0^2}{2} \left(\frac{R^2 \cos^2 \theta}{r^2} - 1 \right)$$

$r \rightarrow$ quadratic

$$\underline{r_{\max} + r_{\min}} = \text{sum of root}$$

Semi major axis = $\left(\frac{r_{\max} + r_{\min}}{2} \right)$

Check $\rightarrow \left(\frac{gR^2}{2gR - v_0} \right) \underline{\text{Ans}}$

GRAVITATION

A Satellite revolving around earth in circular orbit.
of radius r with velocity \underline{v}_0 .

A particle is projected from the satellite in forward direction with velocity $(\sqrt{\frac{5}{4}} - 1)v_0$.

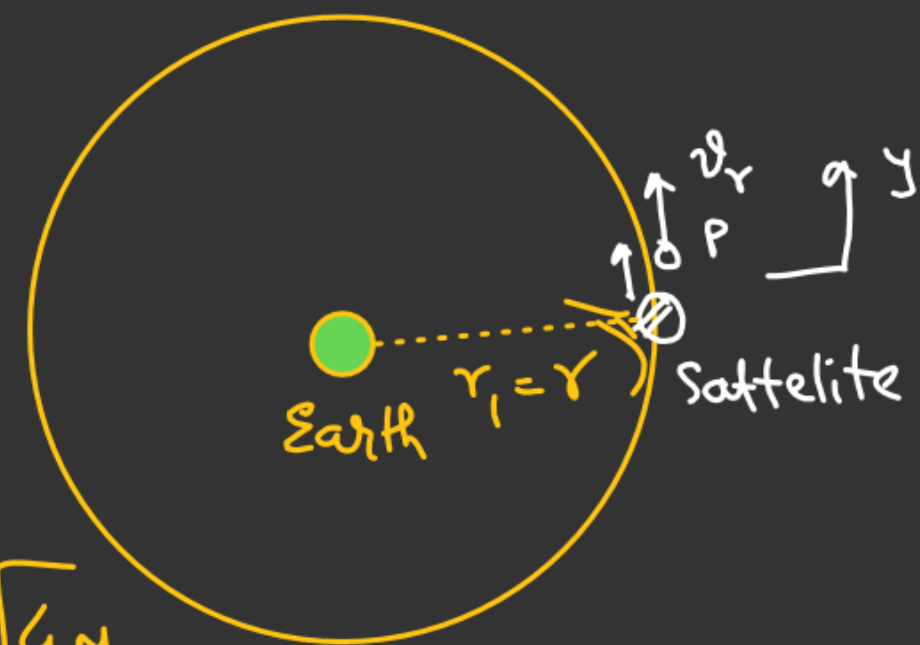
Find max and min distance of the particle w.r.t earth during its subsequent motion

Solⁿ

Absolute velocity of particle

$$\begin{aligned}\vec{v}_{P/E} &= \vec{v}_{P/\text{satellite}} + \vec{v}_{\text{satellite}/E} \\ &= (\sqrt{\frac{5}{4}} - 1)v_0 + v_0 \hat{j} \\ &= \left(\sqrt{\frac{5}{4}} v_0\right)\end{aligned}$$

$$v_p > v_0 \Rightarrow (\text{Trajectory is ellipse}) \quad \left(v_0 = \sqrt{\frac{4M}{r}}\right) \checkmark$$



GRAVITATIONA.M.C.

$$v_0 = \sqrt{\frac{GM}{r}} \quad \checkmark$$

$$\sqrt{\frac{5}{4}} v_0 (r) = v_2 r_2 \quad \text{--- (1)}$$

$$-\frac{GMm}{r} + \frac{1}{2} m \left(\sqrt{\frac{5}{4}} v_0 \right)^2 = -\frac{GMm}{r_2} + \frac{1}{2} m v_2^2 \quad \text{--- (2)}$$

$$\Downarrow$$

$$r_2 = \left(\frac{5r}{3} \right), \quad \text{--- (1)}$$

