

Q1

a) Find total ΔV for transfer

given:

$$\begin{aligned} \mu_{\text{sun}} &= 1.3271 \cdot 10^{20} \text{ km}^3/\text{s}^2 & \text{radius of mercury: } 2440.5 \text{ km} \\ \mu_{\text{mer}} &= 2.2032 \cdot 10^4 \text{ km}^3/\text{s}^2 & \text{radius of saturn: } 60,268 \text{ km} \\ \mu_{\text{sat}} &= 1.2671 \cdot 10^8 \text{ km}^3/\text{s}^2 \\ m_{\text{sun}} &= 1.9885 \cdot 10^{30} \text{ kg} & r_{\text{mer sun}} = 57,909 \cdot 10^6 \text{ km} \\ m_{\text{mer}} &= 3.3010 \cdot 10^{23} \text{ kg} & r_{\text{sat sun}} = 1432,041 \cdot 10^6 \text{ km} \\ m_{\text{sat}} &= 5.6832 \cdot 10^{26} \text{ kg} \end{aligned}$$

from
NASA
fact sheet
NSSDCA

$$r_{\text{craft mer}} = 400 \text{ km} + \text{planet radius} = 2,840.5 \text{ km}$$

$$r_{\text{craft sat}} = 10000 \text{ km} + \text{planet radius} = 70,268 \text{ km}$$

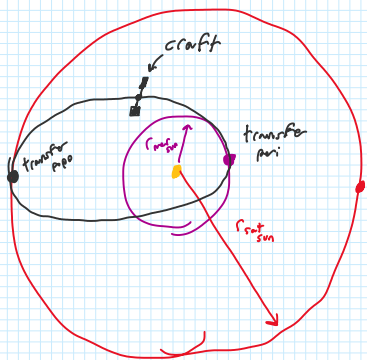
• start solving:

$$V_{\text{mer}} = \sqrt{\frac{\mu_{\text{sun}}}{r_{\text{mer sun}}}} = \sqrt{\frac{1.3271 \cdot 10^{20}}{57,909 \cdot 10^6}} = \underline{\underline{47.8721 \text{ km/s}}}$$

$$V_{\text{sat}} = \sqrt{\frac{\mu_{\text{sun}}}{r_{\text{sat sun}}}} = \sqrt{\frac{1.3271 \cdot 10^{20}}{1432,041 \cdot 10^6}} = \underline{\underline{9.6267 \text{ km/s}}}$$

$$V_{\text{craft, initial mer}} = \sqrt{\frac{\mu_{\text{mer}}}{r_{\text{craft mer}}}} = \sqrt{\frac{2.2032 \cdot 10^4}{2,840.5}} = \underline{\underline{2.785 \text{ km/s}}}$$

$$V_{\text{craft, final sat}} = \sqrt{\frac{\mu_{\text{sat}}}{r_{\text{craft sat}}}} = \sqrt{\frac{1.2671 \cdot 10^8}{70,268}} = \underline{\underline{42.4650 \text{ km/s}}}$$



$$V_{\text{transfer peri}} = \sqrt{2 \left(\frac{\mu_{\text{sun}}}{r_{\text{mer sun}}} - \frac{\mu_{\text{sun}}}{(r_{\text{mer sun}} + r_{\text{sat sun}})} \right)} = \sqrt{2 \left(\frac{1.3271 \cdot 10^{20}}{57,909 \cdot 10^6} - \frac{1.3271 \cdot 10^{20}}{(57,909 \cdot 10^6 + (1432,041 \cdot 10^6))} \right)} = \underline{\underline{66.3726 \text{ km/s}}}$$

$$V_{\text{transfer apo}} = \sqrt{2 \left(\frac{\mu_{\text{sun}}}{r_{\text{sat sun}}} - \frac{\mu_{\text{sun}}}{(r_{\text{mer sun}} + r_{\text{sat sun}})} \right)} = \sqrt{2 \left(\frac{1.3271 \cdot 10^{20}}{1432,041 \cdot 10^6} - \frac{1.3271 \cdot 10^{20}}{(57,909 \cdot 10^6 + 1432,041 \cdot 10^6)} \right)} = \underline{\underline{2.6840 \text{ km/s}}}$$

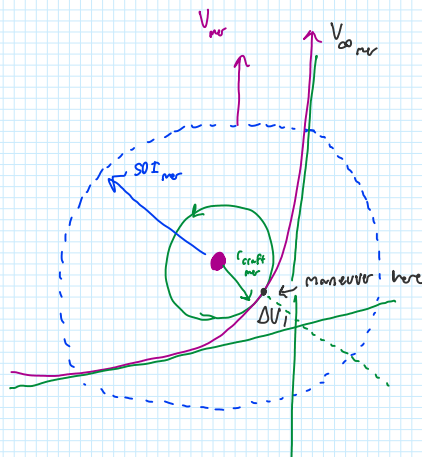
• now zoom in on mercury:

$$V_{\infty \text{ mer}} = V_{\text{transfer peri}} - V_{\text{mer}} = 66.3726 - 47.8721 = \underline{\underline{18.5006 \text{ km/s}}}$$

$$V_{\text{hyperbola, esc mer}} = \sqrt{2 \left(\frac{\mu_{\text{mer}}}{r_{\text{craft mer}}} + \frac{V_{\infty \text{ mer}}^2}{2} \right)} = \sqrt{2 \left(\frac{2.2032 \cdot 10^4}{2840.5} + \frac{(18.5006)^2}{2} \right)} = \underline{\underline{18.9152 \text{ km/s}}}$$

$$\Delta V_1 = V_{\text{hyperbola, esc mer}} - V_{\text{craft, initial mer}} = 18.9152 - 2.785 = \underline{\underline{16.1301 \text{ km/s}}}$$

$$SOI_{\text{mer}} = r_{\text{mer}} \left(\frac{r_{\text{mer}}}{r_{\text{sun}}} \right)^{2/5} = 57,909 \cdot 10^6 \left(\frac{3.3010 \cdot 10^{23}}{1.9885 \cdot 10^{30}} \right)^{2/5} = \underline{\underline{1.1218 \cdot 10^5 \text{ km}}}$$



$$\bullet SOI_{mer} = r_{mer} \left(\frac{m_{mer}}{m_{sun}} \right)^{2/5} = 57.909 \cdot 10^6 \left(\frac{5.3010 \cdot 10^{25}}{1.9985 \cdot 10^{30}} \right)^{2/5} = \boxed{1.1218 \cdot 10^5 \text{ km}}$$

↑
since assuming
circ orbit, $a_{mer} = r_{mer}$

• Now zoom in on saturn:

* Assuming perapsis of hyperbola is
SAME AS PARKING ORBIT!!!

$$\bullet V_{\infty sat} = V_{sat} - V_{transfer Apo} = 9.6267 - 2.6840 = 6.9427 \text{ km/s}$$

$$\bullet V_{hyperbola_{esc sat}} = \sqrt{2 \left(\frac{\mu_{sat}}{r_{craft sat}} + \frac{V_{\infty sat}^2}{2} \right)} = \sqrt{2 \left(\frac{1.2671 \cdot 10^8}{70268} + \frac{(6.9427)^2}{2} \right)} = \boxed{60.4546 \text{ km/s}}$$

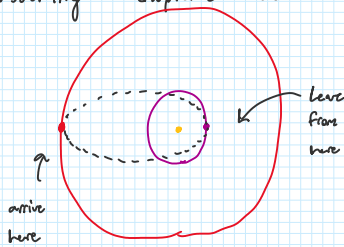
$$\bullet \Delta V_2 = V_{hyperbola_{esc sat}} - V_{craft_{find sat}} = 60.4546 - 42.4650 = \underline{\underline{17.9896 \text{ km/s}}}$$

$$\bullet SOI_{sat} = r_{sat} \left(\frac{m_{sat}}{m_{sun}} \right)^{2/5} = (1432.041 \cdot 10^6) \left(\frac{5.6832 \cdot 10^{26}}{1.9985 \cdot 10^{30}} \right)^{2/5} = \boxed{5.4640 \cdot 10^7 \text{ km}}$$

$$\bullet \Delta V_{tot} = \Delta V_1 + \Delta V_2 = 16.1301 + 17.9896 = \underline{\underline{34.1197 \text{ km/s}}}$$

• FIND TOF:

assuming capture as follows:



Then we can assume an ellipse during
heliocentric transfer:

$$\left. \begin{array}{l} r_p = r_{mer} \\ r_a = r_{sat} \end{array} \right\} a = \frac{r_{mer} + r_{sat}}{2} = 7.44975 \cdot 10^8$$

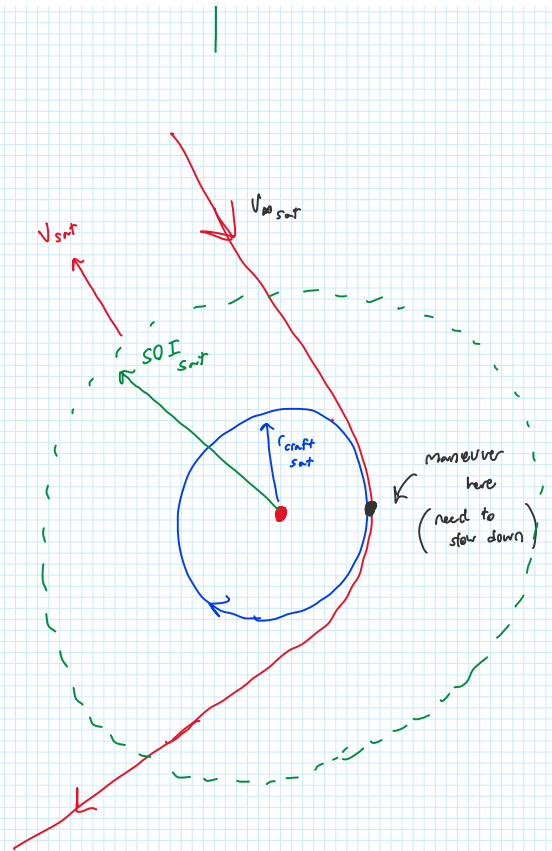
$$r_p = a(1-e) = \frac{r_p}{a} = 1-e \quad e = 1 - \frac{r_p}{a} = \frac{r_{mer}}{a} = 0.9223$$

$E = 180^\circ$ * assuming instant capture

$$t = T = \sqrt{\frac{a^3}{\mu_{sun}}} (E - e \sin(E))$$

$$t = \sqrt{\frac{(7.44975 \cdot 10^8)^3}{1.3271 \cdot 10^{20}}} \left(\pi - 0.9223 \sin(\pi) \right)$$

$$\boxed{t = 1.7535 \cdot 10^6 \text{ seconds}}$$



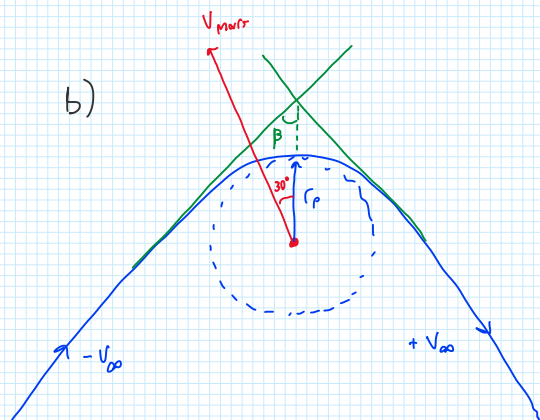
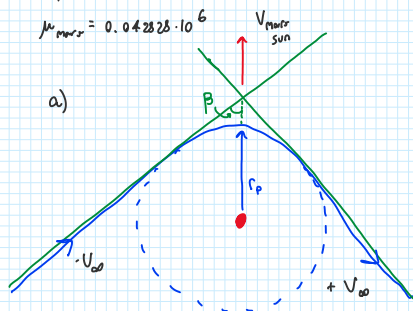
Q2 :

Given:

$$t = 1.2$$

$$r_p = 5380$$

$$\mu_{\text{mars}} = 0.042828 \cdot 10^6$$



c)

