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```
%Section - 01
%Aero 446 Quiz 2: 5/15/25
```

Workspace Prep

Variables

```
Re = 6378; %km
Se = 1366; %w/m^2
mu = 398600;
```

Problem 1

```
% This is on paper
```

Problem 2

```
--- Question 2 With A DOD of 50% the battery can support 100Wh of power in 1 hour
```

Problem 3

```
% summer solstice means beta angle = earth tilt, beta angle is the
% projection of the sun vector onto orbital plane
beta = deg2rad(23.5);
% assume all other components of power generation create a coeff of 1
A = 1;
Pt = (1)*A*cos(0); %power for tracking array

syms A2
eq = Pt == (1)*A2*cos(beta); %power for non tracking array
soln = solve(eq,A2);
A2 = double(soln);

diff = (A2-A)/A;
disp('--- Question 3')
disp(['An percent increase in area of ', num2str(diff*100), '% is required'])
disp('')
--- Question 3
An percent increase in area of 9.0441% is required
```

Problem 4

```
alt = 1111;
r = Re + alt;
beta = deg2rad(30); %assume beta angle of 30 degrees (units are radians)
p = (2*pi)*sqrt((r^3)/(mu)); %period in seconds
ph = p/3600; %period in hrs
inside = ((((Re/r)^2) - (sin(beta)^2))^0.5) / (cos(beta));
fe = (1/pi)*asin(inside); %fraction of orbit spent in eclipse
DOD = 0.5; % 50% depth of discharge
n = 0.9;
td = ph*fe; %time of discharge (time of orbit spent in eclipse)
tc = ph-td; %time the s/c is charging / in the sun
pi = 20; %W when imaging
pr = 10; %W when reading
bus = 10; %W
comm = 20; %W
Poute = pi + bus + comm;
Pout = pr + bus + comm;
```

```
bc = (Poute*td)/(DOD*n); %battery capacity in Wh
Pcharge = ((Pout*tc)+(Poute*td)); %amount of power req each orbit

syms Pin
eq2 = Pcharge == Pin*n;
soln = solve(eq2,Pin);
Pin = double(soln);

disp('--- Question 4')
disp(['Battery Capacity with DOD of 50%: ', num2str(bc), 'Wh'])
disp(['Charging power needed for whole orbit: ', num2str(Pin), 'Wh'])
disp(' ')
--- Question 4
Battery Capacity with DOD of 50%: 58.3461Wh
Charging power needed for whole orbit: 85.4619Wh
```

Problem 5

```
% 1) Battery discharge is kept < 100 because discharging the battery to
% extremely low charge states can: potentially damage the battery or other
% s/c components, reduce total charge capacity, or kill the spacecraft not
% allowing it to turn back on
% 2) K-band is often not used because of attenuation in the atmosphere.
% Nitrogen's or Oxygen's ionization energy is right in the same freq as
% K-band which creates charged particles and greatly interferes with the
% signal
% 3) Diodes only allow current to flow in one direction by creating a
% depletion region in the center of two conductors. The P (positive side)
% and n (nuetral side) have a depletion zone in the middle where the
% charges are flipped creating a break in the circuit. As the current
% increases the diode will allow it to flow one direction as the charges in
% the depletion zone flip. However, powering a diode backwards will not
% allow current to flow because of the depletion zone, although you can
% break the diode and force current the wrong way by putting too much power
% through it.
% 4) (on paper)
% 5) Amplifier 2 needs to go first because you want to create the lowest
% noise signal first before it is greatly amplified.
```

Problem 6

```
gaindB = 7; %dB
input = 10; %W

gain = 10^(gaindB/10);

output = gain*input;
```

```
disp('--- Question 6')
disp(['Output Power: ', num2str(output), 'W'])
disp(' ')
--- Question 6
Output Power: 50.1187W
```

Problem 7

```
Vtn = 3.8; %V
Vd = 28; %nominal operating voltage during discharge
% assume discharge cycle is during eclipse
Pout2 = 10; %W
% battery heat dissipation in watts
Pthermal = (Pout2*Vtn/Vd) - Pout2;
disp('--- Question 7')
disp(['Heat Loss: ', num2str(Pthermal), 'W'])
disp(' ')
--- Question 7
Heat Loss: -8.6429W
```

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