



# THÈSE

## En vue de l'obtention du DOCTORAT DE L'UNIVERSITÉ DE TOULOUSE

Délivré par l'**Institut Supérieur de l'Aéronautique et de l'Espace**

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Le 2 décembre 2019

**Optimisation topologique à formalisme Eulérien et Lagrangien  
appliquée à la conception d'un ensemble propulsif**

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Ecole doctorale : **MEGEP - Mécanique, Energétique, Génie civil, Procédés**

Spécialité : **Génie mécanique, mécanique des matériaux**

Unité de recherche :  
**ICA - Institut Clément Ader**

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# Acknowledgments

I would, first of all, extend my gratitude to Dr. Norato and Dr. Toropov for accepting to review this thesis and Dr. Allaire and Dr. Bartoli for agreeing to be part of the jury. This research work being a CIFRE thesis gave me the opportunity to work both in an academic environment (ISAE and ICA Toulouse) and in a company (Airbus Operations S.A.S.). On the academic side, I would like to thank both my advisors Dr. Christian Gogu and Dr. Joseph Morlier. They trusted my research and my ideas and provided very useful feedbacks. They made me feel capable of doing great things. Their advice now is part of my attitude at work. Double checking is also part of my life now.

At Airbus, I thank my advisor Rémi Amargier, who leaded my research activity and discussed with me way forward to maximize the impact of my developments among my colleges.

I thank ESYT team at Airbus, my coworkers on the plateau and the researcher in ICA lab for trusting the value of my contribution and for fruitful discussions that helped developing my research. I thank all the ISAE master students and all interns I had the chance to mentor in the last 3 years. They also helped me with my personal and professional development.

I want to extend my thanks to my wife who supported my passion for my research and helped me achieving all my personal and professional goals. I thank also my family for investing in my success and believing in me.

I want also to extend my gratitude to all researchers I have discussed with in conferences that also gave me valid hints for improving my framework among others Dr. Xu Guo, Dr. Weisheng Zhang, Dr. Oded Amir, Dr. Niels Aage, again Dr. Julian Norato, Dr. H Alicia Kim, Dr. Frédéric Duboeuf and Dr. Anders Klarbring.

**Abstract** — The integration of high bypass ratio engines currently represents a huge challenge for aircraft manufacturers. In fact, structural issues are caused by their geometric characteristics: increased fan diameter and decreased core diameter. The engine deformations induced by maneuver loads must be controlled to avoid excessive variations of the clearances between rotating blade tips and the fixed engine casing. These clearances are known as "tip-clearance" and if not controlled can increase the engine specific fuel consumption, reduce operability and increase maintenance costs. The structures integrating the engine under the wing, i.e. pylon, engine mounts and nacelle, impact the way in which the engine deforms under maneuver loads. Therefore these structures can be designed to reduce tip clearances (and consequently specific fuel consumption) variations. An aircraft manufacturer, has very limited control on engine design, but has main control over the integration of the engine to the rest of the aircraft and seeks novel architectures for this integration.

Given this context, the main goal of this PhD consists in developing numerical tools for exploring novel architectures for engine to wing integration by controlling engine deformations and thus fuel consumption variations. Topology optimization was chosen as a framework for the exploration of such novel designs. Topology optimization is a mathematical method that optimizes material layout within a given design space, for a set of loads, boundary conditions and constraints, with the goal of maximizing the performance of the system. In our context, this technique can also help recognizing integration key design choices affecting tip-clearance variations.

To use such an approach, a structural model of both engine and design space are required. The engine model can be considered as an input to the aircraft manufacturer. Usually it is delivered by the engine manufacturer as a finite element model in a commercial software such as Nastran or Abaqus. The engine model was integrated in the developed framework by the use of a superelement strategy that avoids implementation complexity and discrepancies in finite element responses.

The topology optimization of the design space between the engine and the wing was considered, including stress constraints and performance criteria (fuel consumption variations). The proposed approach allows the control of both fuel consumption variations and sizing when architecture choices are made. In this thesis we make the distinction between Eulerian and Lagrangian approaches to topology optimization. On one hand, in Eulerian approaches the solution is described by a density field that identifies material and void regions in the design space. On the other hand Lagrangian approaches solutions are described as assemblies of simple geometric primitives that can be combined to achieve complex designs. In this thesis both are investigated as they can contribute with complementary solutions.

To avoid Eulerian approaches inherent issues, density filtering was considered in this PhD and, to reduce the computational burden required to build the filtering matrix, a multigrid strategy was proposed. The Unified Aggregation Relaxation approach was considered to make singular optima accessible and efficiently deal with a large number of stress constraints. Finally, to efficiently apply constraints in the optimization problem the adjoint evaluation of gradients of stress and performance constraints was derived.

In the context of Lagrangian topology optimization, we proposed a method generalizing existing techniques (Geometry Projection, Moving Morphable Components and Moving Node Approach). This approach was validated on classic 2D numerical examples (Short cantilever

beam and L-shape). It was then extended to the optimization of the engine to wing attachment.

The results obtained with both Eulerian and Lagrangian approaches on a generic engine integration design usecase are consistent with modeling hypothesis and illustrate novel opportunities for original designs of engine to wing attachments. The developed framework thus provides innovative tools for exploring new designs for improving the engine-to-aircraft integration.

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**Keywords:** Topology Optimization, Finite Element Analysis, Mesh tying, Geometric Multigrid, Geometry Projection, Moving Morphable Components (MMC), Moving Nodes Approaches (MNA).

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**Résumé** — L'intégration de moteurs turbofan caractérisés par un très grand taux de dilution représente aujourd'hui un énorme défi technique pour les fabricants d'avion. En effet, des problèmes structuraux sont causés par leur caractéristiques géométriques, c.à.d. un diamètre de la soufflante élargi et un diamètre de la turbine proprement dite réduite. Les déformations du moteur induites par les charges en manœuvres de l'avion doivent être contrôlées pour éviter de trop grandes variations des jeux entre les parties tournantes et parties fixes du moteur appelées "tip clearance". Ces jeux, s'ils ne sont pas contrôlés, peuvent engendrer une surconsommation du moteur, limiter son opérabilité, ou augmenter les couts de maintenance. Les structures qui intègrent le moteur sous la voilure, c.à.d. le mât, les attaches moteur et la nacelle, peuvent impacter la façon dont le moteur se déforme sous l'action des charges dû aux manœuvres de l'avion. Pour cette raison ces structures peuvent être conçues pour réduire les variations de "tip clearance" et par conséquent les variations de consommation du moteur. Le constructeur d'avions, a une influence limitée sur la conception du moteur, mais a le contrôle sur l'intégration du moteur au reste de l'avion et cherche des nouvelles architectures pour cette intégration. Dans ce contexte, l'objectif principal de la thèse est de développer des outils numériques pour l'exploration de nouvelles architectures pour l'intégration du moteur à la voilure tout en gardant sous contrôle les déformations du moteurs et donc la variation de consommation de carburant. L'optimisation topologique a été sélectionnée comme outil principal d'exploration pour des tels conceptions innovants.

Par définition, l'optimisation topologique est une méthode mathématique qui cherche à trouver le meilleur arrangement de matière dans un espace de conception donné pour des chargements, des conditions limites, des contraintes et des objectifs donnés. Dans notre contexte, cette technique permet de reconnaître quels sont les choix architecturaux qui impactent le plus les variations de "tip-clearance". Pour utiliser une telle approche un modèle structure à la fois du moteur et de l'espace de conception sont nécessaires. Le modèle moteur peut être considéré comme un input pour le fabricant d'avions. En effet, celui-ci est souvent livré par le motoriste comme un modèle élément finis dans des logiciels commerciaux comme Nastran ou Abaqus. Pour considérer ce modèle dans notre environnement de développement nous avons utilisé des superelements. Cela évite une très grande complexité d'implémentation et des différences dans les réponses d'intérêt. L'optimisation topologique de l'espace de conception entre le moteur et la voilure a été considérée, en incluant des contraintes de stress admissible et critères de performances (variations de consommation de carburant). L'approche qu'on propose permet de contrôler la variation de consommation et le dimensionnement des structures pour une architecture donnée. Dans ce manuscrit, nous mettons en avant deux formalismes : l'approche Eulérienne et l'approche Lagrangienne. Dans la première famille, le concept est décrit à l'aide d'un champ de densité qui identifie les zones pleines et vides dans la région de conception. Les approches Lagrangiennes décrivent, elles, le concept comme un assemblage de composants caractérisés par une géométrie simple. Nos développements méthodologiques s'appuient sur ces deux formalismes pour résoudre le problème industriel de conception d'architecture du système propulsif (ensemble mat réacteur et attaches moteur), étant donnée la complémentarité des solutions proposées par ces approches. Pour éviter problèmes numériques associés avec les approches Eulériennes, le filtre de densité a été considéré dans ce travail et pour réduire le nombre d'opérations associé avec la construction de la matrice de filtrage, une stratégie multi-maillage a été proposée. Optimiser des struc-

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tures composées de millions d'éléments présente aussi un défi important dans le traitement des contraintes en stress. Celui a été résolu en utilisant une stratégie unifiée d'agrégation et de relaxation. Les dérivées de chaque réponse considérée dans le problème d'optimisation topologique ont été calculées analytiquement avec la méthode adjointe. Nous avons aussi proposé, dans le cadre de l'optimisation topologique Lagrangienne, une méthode généralisant les techniques existantes (Projection Géométrique, MMC, MNA). Cette approche a été validée sur des exemples académiques classiques de la littérature (Poutre encastrée 2D, Poutre en L 2D). Enfin, l'environnement d'optimisation topologique développé a été testé sur un exemple de conception d'intégration de moteur. La méthodologie développée fourni ainsi des outils innovants pour l'exploration de nouveaux conceptions et pour améliorer l'intégration du moteur à l'avion.

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**Mots clés :** Optimisation topologique, Éléments finis, Interface de maillage, méthode multi-grille, projection géométrique MMC (Moving Morphable Components), MNA (Moving Nodes Approach).

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# Nomenclature

<b>PPS</b>	<i>Power-plant Systems</i>
<b>TSFC</b>	<i>Thrust Specific Fuel Consumption</i>
<b>FEM</b>	<i>Finite Element Model</i>
<b>FEA</b>	<i>Finite Element Analysis</i>
<b>MMC</b>	<i>Moving Morphable Components methods</i>
<b>MNA</b>	<i>Moving Node Approach</i>
<b>GP</b>	<i>Geometry Projection</i>
<b>GGP</b>	<i>Generalized Geometry Projection</i>
<b>WEM</b>	<i>Whole Engine Model</i>
<b>DOF</b>	<i>Degree Of Freedom</i>



# Introduction

## Résumé

Les émissions de  $CO_2$  anthropogènes contribuent au réchauffement climatique par des phénomènes physiques aujourd’hui bien compris [Change et al., 2007]. Le secteur du transport aérien est en particulier responsable de 2 % du total des émissions provoquées par l’activité humaine ( [ICAO, 2016]) mais ce chiffre est destiné à augmenter dans les prochaines années ( [Terrenoire et al., 2019]). Pour cette raison des mesures concrètes pour la réduction des émissions de  $CO_2$  s’appliqueront aux nouveaux avions à partir de 2020 et aux avions en production à partir de 2019 [National Academies of Sciences et al., 2016]. Si l’on classe les avions par leurs capacités on peut s’apercevoir que les plus grands consommateurs de carburant sont les avions avec une capacité supérieure à 100 passagers [Yutko, 2011, Epstein and O’Flarity, 2019]. Cela implique que l’effort de recherche doit être dirigé sur la réduction des émissions de ces dispositifs. Ces avions sont souvent propulsés par des moteurs thermiques du type turbofan. Dans ces moteurs la poussée est fournie par une soufflante qui est entraînée dans sa rotation par une turbine. Pour réduire leur consommation de carburant, l’architecture de ces systèmes évolue vers des concepts à très haut taux de dilution (UHBR). Cela demande une réduction du diamètre du carter de la partie core (turbine proprement dite) et une augmentation du diamètre de la soufflante. L’intégration d’un tel moteur représente un défi technologique majeur. En effet due à sa géométrie les déformations moteur seront plus difficiles à contrôler en particulier la variations des jeux entre partie tournante et partie fixe du moteur. Ces jeux appelés "tip-clearance" peuvent en effet changer à cause du chargement appliqué au moteur à cause des manœuvres de l’avion. Cela a différentes conséquences néfastes sur son fonctionnement, entre autre l’augmentation de leur consommation de carburant [Lattime and Steinetz, 2002]. Ces moteurs sont souvent montés sous la voilure par un mât et couvert par une coque aérodynamique appelée nacelle. Les deux contribuent à la performance finale par différents phénomènes. Dans cette thèse on s’intéresse à l’impact que les raideurs de la structure primaire du mât et de la nacelle ont sur la performance moteur par leur impact sur les déformations moteurs et donc sur les "tip-clearance". Pour avoir une quantification de ces impacts la simulation par éléments finis a été utilisée. De plus pour pouvoir identifier des solutions innovantes, l’optimisation topologique a été considérée. Le but de cette optimisation structurale est d’identifier des structures ayant des performances optimales sous des chargements définis et ayant un budget en masse défini. Dans cette thèse on a en particulier considéré 2 familles d’approches: les approches dites Eulériennes et Lagrangiennes [Zhang et al., 2016d] selon la description faite de la solution. Si la solution est décrite par des champs de fonctions qui indiquent la présence ou l’absence de matière dans chaque point de l’espace de design alors on parle d’approches Eulériennes. D’autre part si la solution est décrite par une assemblage de formes élémentaires qui

peuvent changer de positions et de dimensions alors on parle d'approches Lagrangiennes. Les deux ont des avantages et des inconvénients qui sont étudiés dans cette thèse. Pour réaliser cette étude et l'appliquer à un contexte industriel un cadre d'analyse par éléments finis et d'optimisation topologique a été développé. Pour cela différents défis techniques ont été soulevés:

1. Le traitement des modèles éléments finis industriels de moteur.
2. Le géométrie de région de design quelconque.
3. La communication de maillages incohérents.
4. L'efficacité et la scalabilité des algorithmes considérés.
5. L'implémentation des contraintes en stress dans l'optimisation topologique.
6. L'implémentation des approches Lagrangiennes.
7. L'introduction de composantes géométriques spécifiques au problème industriel.

## Industrial context

*Aviation CO<sub>2</sub> reduction challenge*

Energy production and transportation nowadays rely on fossil fuel combustions such as coal, natural gas and oil. As a consequence, human activity contributes to the release of carbon dioxide ( $CO_2$ ) and other greenhouse gases like methane, nitrous oxide and fluorinated gases. The physical phenomena that make the link between global warming and greenhouse gases anthropogenic emissions are well understood [Change et al., 2007]. Since the first industrial revolution,  $CO_2$  has been the major contributor to global warming. To face this situation, governments and private companies of all industrial sectors started working for the reduction of  $CO_2$  emissions. In 2016, the International Civil Aviation Organization (ICAO) recalled that the aviation sector ‘accounts for under 2 % of the world’s annual  $CO_2$  emissions’ ([ICAO, 2016]). As commercial aviation keeps growing in term of revenue-passenger miles and cargo ton miles,  $CO_2$  emissions are expected to increase [Terrenoire et al., 2019] therefore actions to reduce them are urgent and need to be considered in the present. It is in fact expected that a new fuel economy standard will be incorporated in national rules to reduce both aircraft emissions and noise. It will apply to new aircraft design in 2020 as well as to in-production type in 2023 [National Academies of Sciences et al., 2016].

Commercial aircraft can be organized in the following families:

- General aviation: fewer than 6 passengers
- Commuter: fewer than 20 passengers
- Regional: 30-100 passengers
- Single-aisle: 100-200 passengers

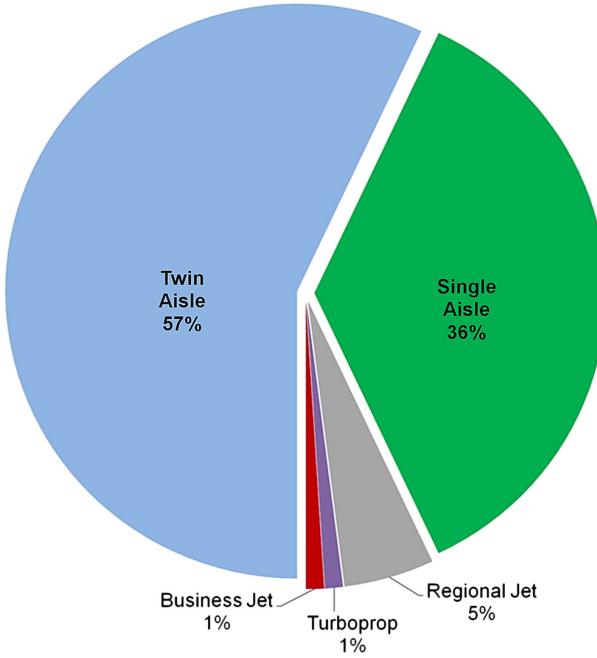


Figure 1: Global civil aviation fuel consumption. Source: [Yutko, 2011, Epstein and O'Flarity, 2019]

- Twin-aisle: more than 200 passengers

In figure 1 it is reported the distribution of global fuel consumption for each class [Yutko, 2011]. It can be observed that more than 90 % of the  $CO_2$  emissions from global commercial aircraft operations are generated by large aircraft. That is why single-aisle and twin-aisle are subjected to large research investments for their fuel consumption reduction. These families of aircraft are often propelled by turbofan engine (c.f. figure 2). This gas turbine engine is composed by a fixed casing and by several spools. Airflow passing through the fan and then in compressors, combustion chamber and turbines follows a Brayton cycle. Thanks to the excess of power produced by this cycle, the fan provides engine thrust. Engines on current turbofan-powered commercial aircraft are mounted on pylons, which keep the engines off the wings or fuselage in order to isolate the engine and airframe aerodynamic characteristics (c.f. figure 3). The pylons must transmit the thrust loads from the engine to the airframe as well as convey all fluid and electrical interconnections. Among pylon substructures (c.f. figure 4), primary structure transfers engine loads to the wing. This is ensured by a system of connections between pylon and engine or wing (so called engine and wing mounts respectively) and by the pylon torque box that joins these two. The engines are enclosed by fairings known as nacelles (c.f. figure 5), which contain many of the subsystems important to the operation of the aircraft such as the electrical generators. The nacelles also serve other purposes, including aerodynamic fairing of the engine, conditioning of airflow into the engine, thrust reversing, and noise attenuation. The aerodynamic, structural, and subsystem integration of the engine and nacelle with the airframe are important to determine aircraft performance and optimum engine characteristics such as propulsor diameter and fan pressure ratio.

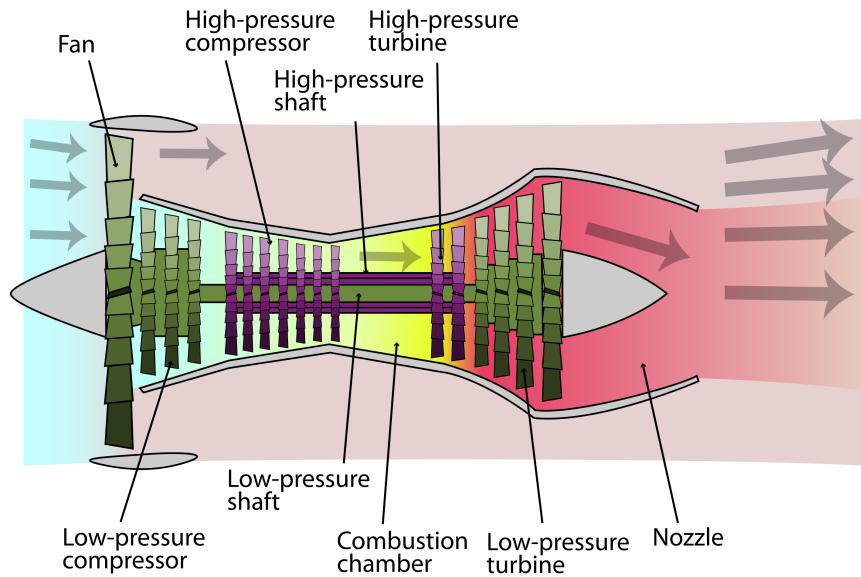


Figure 2: Turbofan schematic representation. SOURCE: Wikipedia

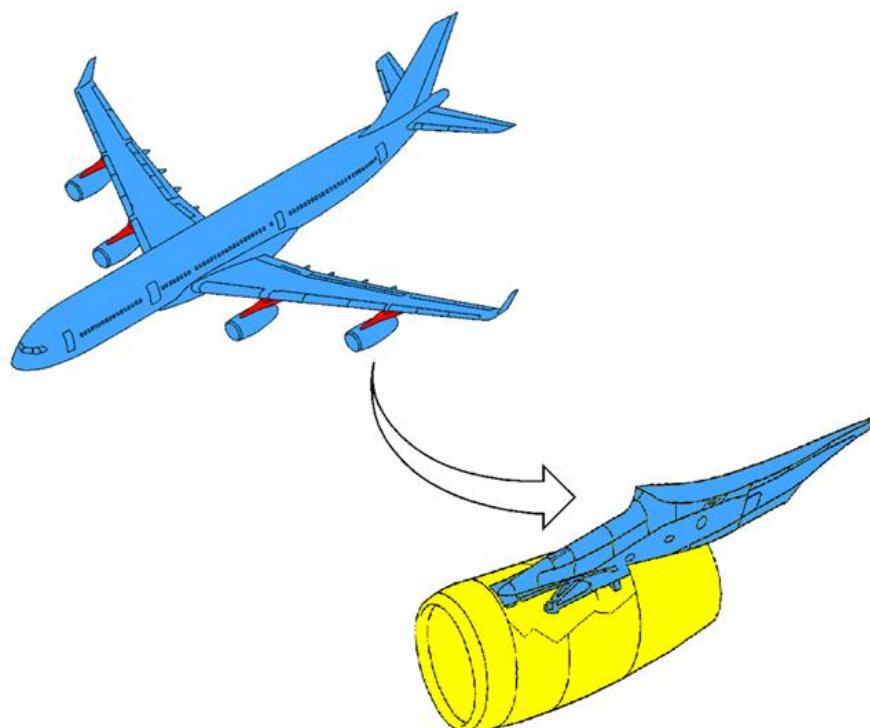


Figure 3: Commonly adopted pylon position in large turbofan propelled aircraft. Source: [Cros, 2013]

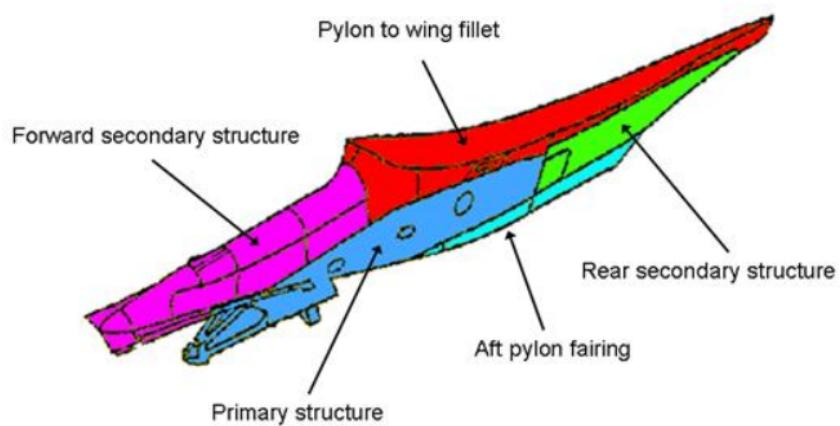


Figure 4: Pylon structure (A340 example). Source: [[Cros, 2013](#)]

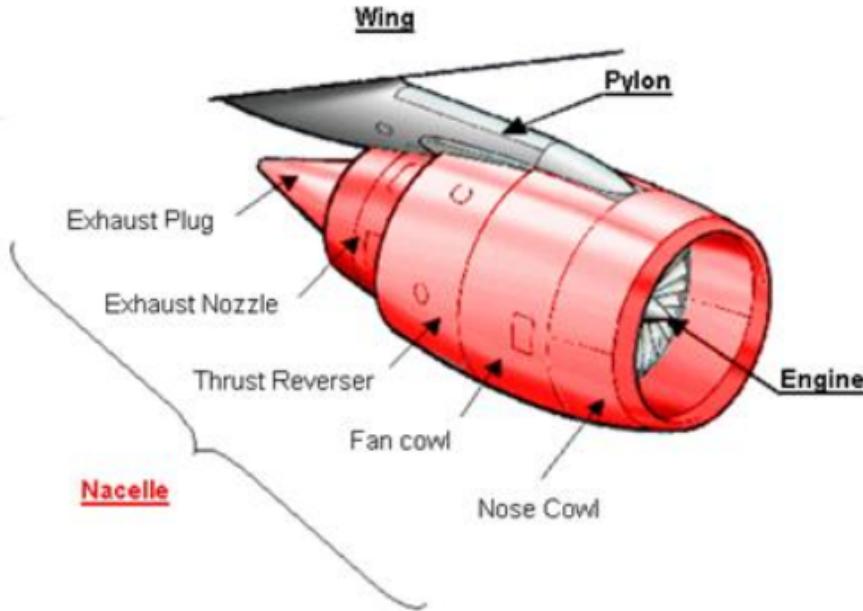


Figure 5: Nacelle subsystems. Source: [Cros, 2013]

*Increasing the Bypass Ratio* To improve the existing propulsion efficiency several technologies are currently investigated [National Academies of Sciences et al., 2016]. One of those consists in increasing the so called bypass ratio. This is defined as the ratio between the airflow passing through the fan and then ejected from the fan nozzle (which mainly contributes to engine thrust) and the airflow passing through the combustion chamber (responsible for power generation). Increasing turbofan bypass ratio has the following consequences on design and performances:

- Fan diameter needs to be increased. As a consequence, the propulsion efficiency is also improved. In fact for the same level of thrust provided by the fan, outlet air velocity can be reduced, consequently reducing losses and noise.
- Core diameter needs to be decreased. This is because aircraft propulsive efficiency is improved and that means that less power is demanded for the same level of engine thrust. Moreover this is also needed to improve thermodynamic efficiency when the temperature and pressure in the first stage of turbine are increased. [National Academies of Sciences et al., 2016].

*Tip-clearance variation challenge* Engine and integration design are challenged by these drivers for several reasons. Among others, such designs have to deal with larger engine deformations that can deteriorate the control of stator to rotor blade tip clearances, defined as the radial gap between the rotor blade tips and the engine casing. In fact a larger engine will be subjected to larger loads and eventually to larger deformations. On top of that the center of gravity of the engine is moved forward possibly amplifying the effect of inertial loads. An appropriate sizing needs to be considered in order to avoid excessive closures or openings. An abradable coat material is often applied to the engine casing. This sacrificial material is often milled by the rotor

due to engine deformations that can close these gaps. When this happens, the aerodynamic performance of the stage is reduced, increasing the overall engine specific fuel consumption. Moreover, to keep the same level of thrust, more fuel needs to be injected in the combustion chamber consequently increasing the temperature of outlet gas. Engine material can then be subjected to high temperature corrosion phenomena that reduce the engine time on wing [Lattime and Steinetz, 2002]. In compressor stages these gaps have even more severe consequences since surge margin is reduced. This means that not only fuel consumption is affected, but also safety and engine operability [Benito et al., 2008]. Also in the fan stage the openings reduce the propeller efficiency further impacting the specific fuel consumption.

Tip clearance variations in both compressor and turbine stages can be due to both engine loads (propulsion induced) and flight loads. Engine loads include centrifugal, thermal, internal engine pressure, and thrust loads. Flight loads include inertial (gravitational), aerodynamic (external pressure), and gyroscopic loads. Engine loads can produce both axisymmetric and asymmetric clearance changes (see Figures 6 and 7). Flight loads produce asymmetric clearance changes [Lattime and Steinetz, 2002]. Axisymmetric variations are mainly driven by

*Tip-clearance drivers*

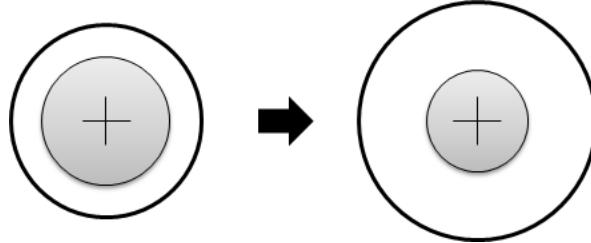


Figure 6: Axisymmetric clearance variations

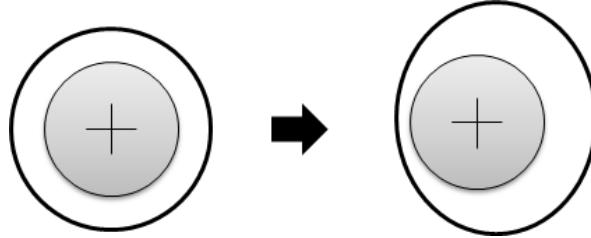


Figure 7: Asymmetric clearance variations

engine rotor and stator sizing and by operating conditions. Nacelle and pylon are both linked to the engine. This means that their stiffness can have an impact on loads passing through each interface with the engine, therefore on the engine deformations. As a direct consequence pylon and nacelle design has an impact on tip clearance variation. [Lattime and Steinetz, 2002]. This contribution to the overall tip clearance variation in classic turbofan engine represents about a third of the total variation induced by other sources [Lattime and Steinetz, 2002]. Nevertheless the relative importance of such contribution is still to be quantified for Ultra High Bypass Ratio (UHBR) architectures (Bypass ratio of the order of 30).

The investigations that was conducted here, consist in studying how design choices made A design on:  
problem

- Engine outer casing main structure
- Engine to wing load path
- Nacelle and pylon main structure

affect engine tip clearance variations in the context of UHBR architecture. For the aforementioned considerations such designs could be beneficial for both  $CO_2$  emissions, engine life and safety. In the determination of such a design one should be able to estimate the performance, in terms of tip clearance variation control and select a design considered 'best' among others.

Using only engineering considerations in the context of such complex phenomena could possibly be misleading or insufficient. For this reason an automated procedure, based on Simulation optimization algorithms in combination with a simulation model are proposed. A mathematical description of each configuration is then required and corresponding models should be available. In particular, structural optimization approaches deal with mechanical responses and often try to achieve the lightest or the stiffest design responding to some strength requirements on the solution.

Depending on the design mathematical description, the optimal solution is more or less driven by the modeling choice made on the solution. In the industrial problem considered

Why by this study, assumptions made on the connectivity between pylon and engine can drive the Topology solution to configurations that one cannot ensure to be optimal. For this reason a way of Optimiza- considering a large spectrum of possible solutions is by addressing the problem in a topology tion? optimization framework.

## Topology Optimization approaches

Topology optimization is a powerful tool that can be used to explore structural design. The inputs for topology optimization are the geometry of the design zone (i.e. a region in the space delimitating the structure), boundary conditions and materials. A structural model (often a Finite Element model) is generated to compute each configuration's responses. The formulation can then be stated, in the form of a non-linear constrained optimization problem. This means that a model response is selected as objective to be minimized or maximized and other responses are constrained. For what concern the design variables, a general distinction can be made at this level:

- The design variables are associated to the presence or the absence of material, at a given point/element in the space. These approaches are referred to as Eulerian in this work [Zhang et al., 2016d], in reference to the continuum mechanics and the representation of displacements. In this first family one can find more common density based

approaches [Bendsøe, 1989, Zhou and Rozvany, 1991, Bendsøe and Sigmund, 1995], Level Set approach [Wang et al., 2003, Allaire et al., 2004] and , evolutionary approaches [Xie and Steven, 1993, Xia et al., 2018].

- The design variables are associated with geometric properties of features that are assembled to obtain the final solution. This second family are referred to as Lagragian or explicit approaches [Zhang et al., 2016d]. Classic examples are the Moving Morphable Components approach (MMCs) [Guo et al., 2014, Guo et al., 2016, Zhang et al., 2016c, Zhang et al., 2017d], the Moving Morphable Voids (MMVs) [Zhang et al., 2017c], the Geometry Projection Method (GP) [Bell et al., 2012, Norato et al., 2015, Zhang et al., 2016a], the Method of Moving Morphable Bars (MMB) [Hoang and Jang, 2017] and the Moving Node Approach (MNA) [Overvelde, 2012].

In both cases design variables are constrained by lower and upper bounds. Once that the optimization problem is stated, gradient based optimization algorithms are typically employed to improve an initial guess design. When a solution is determined, in both cases the solution has to be interpreted in terms of geometry, manufacturing technology and advanced design considerations. Further optimizations (shape and sizing) are often casted to obtain sized designs that respect reserve factors (i.e. safety factors). In [Zhu et al., 2016] one can find an up to date review of the most promising applications of topology optimization to aerospace structures. Topology optimization has shown its advantages for the design of aircraft parts like the wing internal primary structure [Eves et al., 2009, Aage et al., 2017], the wing-box, fuselage [Niemann et al., 2013, Singh et al., 2016] and of the pylon [Remouchamps et al., 2011, Xue et al., 2012].

## PhD goals

As aforementioned the load path between engine and wing, is a contributor to engine performance and in particular fuel consumption. Note that there are larger contributors to the engines' performances which are exclusively dependent on the engine manufacturer's design choices. However the engine's attachment to the wing and the resulting load path, is a significant contributor to the performance while being mainly a design choice of the aircraft manufacturer. Physical intuition can be effective in this choice, only for very simple considerations. On the other hand to prevent complex phenomena like engine casing ovalization, physical intuition is often misleading. For this reason simulation should be used as an exploration tool to get rid of bad solutions and select promising configurations. To address such a complex problem, without making too strong hypotheses on the solution's connectivity, a topology optimization strategy is investigated in the present work. In this PhD topology optimization framework is developed, initially intended to use Eulerian approaches and successively adapted for Lagrangian approaches to address this design problem. Eulerian approaches are able to represent organic designs that can be considered as reference solutions, even if not attainable by nowadays manufacturing technologies. Lagrangian approaches on the other hand make an assumption on the basic components that should compose the solu-  
Engine-wing load path determination Research axes

tion. For this reason their power of representation is restrained, but their advantage is their link with an explicit geometry representation. This can both decrease the overall optimization computational effort and the end to end design effort. The following challenges should then be addressed in order to deal with the engine-to-wing attachment design problem:

1. The proposed framework should be able to deal with industrial engine finite element models allowing also the post-processing of tip clearance variations. Such models are often provided in industrial finite element software like Nastran or Abaqus. An efficient interface with such programs should be therefore provided.
2. Complex geometries of the design zone may require at least non uniform meshes. In fact due to aerodynamic needs and also to the kinematic interfacing with the engine and design region models, voxel meshes should be avoided.
3. Finite element simulations including inconsistent mesh interfaces between models should be allowed. The engine model being provided by the engine manufacturer comes in a given discretization. To avoid constraints on design region discretization the provided framework should be able to connect non consistent discretizations.
4. Efficiency and scalability should be implementation drivers as the resolution of 3D topology optimization problems can quickly become prohibitive. In fact reducing finite element mesh size the computational burden associated with both analysis and optimization is increased. The provided framework should therefore employ adapted techniques for both.
5. The possibility to consider stress based topology optimization. In fact most topology optimizations are concerned with structural stiffness. On the other hand the stiffest design is not always the strongest. Dealing with stress problem is challenging and the provided framework should adopt appropriate techniques.
6. The use of Lagragian approaches in topology optimization. A shortcoming of Eulerian approaches is their lack of an explicit control on the final solution geometry, often requiring human intervention and solution simplifications to provide an acceptable design. Lagrangian approaches provide an efficient solution to this issue and should therefore be provided by the proposed framework.
7. The introduction of explicit geometric constraints linked to the product to be designed. In fact another advantage of Lagrangian approaches is the possibility to use constitutive components that can make easier the determination of a manufacturing process for the solution. The proposed framework will allow this investigation.

At the best of the author's knowledge at the beginning of this project there wasn't any industrial or academic framework that could provide a satisfactory answer to all these requirements. For this reason a novel topology optimization framework will be developed in this study. The goal of this work can be resumed in this question:

*Is it possible to identify disruptive design features of engine pylon and mounts architecture*

*that have an impact on tip clearance variations?*

The question was investigated by the development of a novel topology optimization framework that had to provide effective solutions to the aforementioned challenges.

## Layout

The investigation of this work can be divided in 3 large topics that will be covered in this manuscript, the finite element analysis, the Eulerian topology optimization approaches and the Lagrangian topology optimization approaches. As a consequence the remainder of this work is organized as follows:

- **Chapter 1** deals with challenges related finite element analysis, state of the art approaches and proposed solutions are covered in this chapter. The fundamentals of linear static analysis and finite element analysis are reviewed in section 1.1. A representative use case engine finite element Model, built for this work, is presented in subsection 1.2.1. The post processing of tip clearances required to assess a performance analysis is also presented in subsection 1.2.2. In section 1.3 some advanced techniques for the interface of non-consistent meshes (Mortar approach [Bernardi, 1989], Rescaled Localized Radial Basis function interpolation [Deparis et al., 2014] and Internodes [Deparis et al., 2016]) are reviewed. Their shortcomings in terms of accuracy and complexity are treated with the introduction of novel techniques [Coniglio et al., 2018a]. To deal efficiently with an industrial engine model, static condensation and superelement exploitation was reviewed 1.4.1. Finally to efficiently solve the system of balance equations, efficient multigrid preconditioners for iterative solvers are reviewed and benchmarked on a 3D finite element model in subsection 1.4.2.
- **Chapter 2** deals with challenges encountered in structural topology optimization by the use of the Solid Isotropic Material with Penalization (SIMP) approach. Simulation driven design is reviewed in section 2.1 as the mathematical formulation of SIMP topology optimization in section 2.2. Constraint aggregation and relaxation techniques are reviewed to deal with stress based topology optimization formulations in section 2.3. In section 2.4 the SIMP application to pylon and engine mount design is presented.
- **Chapter 3** deals with challenges encountered in structural topology optimization by the use of Lagrangian approaches. Moving Morphable Components, Geometry Projection Method, Moving Node Approach are firstly reviewed in sections 3.2, 3.3 and 3.4 respectively. A new Generalized Geometry Projection approach is proposed as a generalization of existing techniques in section 3.5. Geometric assembly techniques are then reviewed in section 3.6. Analytic derivatives computation is then provided in section 3.7. The application to 2D usecase problems for both stiffness based and stress based formulations are provided in section 3.8 and 3.9 respectively. Finally the application to the engine pylon and mount design of a novel MNA-type approach is considered in section 3.10.

- **Chapter 4** outlines the main conclusions and the perspectives for the future research related to this work.

# Structural analysis by FEM

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## Résumé

Dans le chapitre d'introduction nous avons relevé un besoins et un cahier des charges pour le cadre que l'on souhaite développer. Pour cela un modèle utilisable sera nécessaire pour prédire

- Les déformations d'un moteur intégré soumis à des chargements.
- L'impact de ces déformations sur les "tip-clearance".
- L'impact des variations de "tip-clearance" sur la consommation du moteur.

Dans un premier temps nous rappelons le bases de théorie de la mécanique des solides déformables. Ensuite la méthode des éléments finis est introduite comme moyen de construction de modèle. Une maquette éléments finis de moteur est introduite pour la prédictions des variations de "tip-clearance" et de performance moteur. Pour pouvoir traiter la communication entre maillage incohérent, une étude bibliographique est d'abord présentée et deux nouvelles contributions sont introduites. L'approche WACA essaye de fournir un bon compromis entre précisions, complexité et cout computationnel. L'approche de correction de moments fournie une technique d'amélioration de toutes les techniques existantes en imposant "apriori" la conservation du bilan des moments mécaniques à l'interface entre maillages. Ces techniques ont été implémentées et comparées avec différentes approches disponibles dans la littérature. Étant le modèle de moteur intégré utilisé dans les prochains chapitres pour faire de l'optimisation topologique, son cout d'évaluation doit être contenu le plus possible. Une première approche proposée pour réduire le temps de calcul et pour gérer des modèles moteurs complexes consiste à utiliser des super-éléments. Cela nécessite l'utilisation de matrices fournies par les logiciels commerciaux. Pour cela nous faisons un rappel sur les étapes nécessaires à leur exploitation. Ensuite la méthode itérative du gradient conjugué avec préconditionneur multigrad géométrique est aussi implémentée et testés avec différents smoothers pour réduire les temps de calcul de l'analyse éléments finis du problème en étude.

## 1.1 Finite elements method in structural analysis

Structural analysis is a major part in the design and validation phase in many industrial sectors. Its objective consists in predicting the structural integrity and performance of structures under different phases of the product life cycle. The aircraft industry relies on structural analysis for both safety and performance purposes. A product is in fact the object of several phases of testing before its integration. Experimental testing is the most expensive and time demanding part as it needs the construction of prototypes, with increasing levels of fidelity and cost with the development phase. Moreover sensors, actuators, and testing facilities are needed for the appropriate simulation of the product environment. On top of that, correctly understanding test results requires time, experience and high-skilled profiles. For all these reason, experiments, even if necessary, have to be reduced as much as possible. For this reason numerical simulation is nowadays employed for helping both explaining test results and reducing the overall number of experiences. At the roots of this methods stand a solid knowledge of the physics governing a particular phenomenon that has to be studied. The continuum hypothesis is considered as satisfied when studying structures on length scales much greater than that of inter-atomic distances. The governing equation of continua can

be mathematically described by systems of Partial Differential Equations (PDEs), where the unknowns are fields that represent the quantity of interest that one would like to know in any point of the space. Finite Element Modeling (FEM) is the most popular approximation method used to solve PDE problems in industrial applications. FEMs can in fact provide:

- Models for structures and fluid mechanical behavior.
- Models for the quantification of thermal exchange phenomena.
- Models for the quantification of electromagnetic phenomena.
- Models that account for coupled multi-physics phenomena.

The main goal of the beginning of this chapter is to introduce notations associated with the finite element formulation for the linear elasto-static problem, notations which will be used throughout the rest of the thesis. The optimization responses needed in chapter 2 need the evaluation of both a mechanical finite element model (c.f. subsection 1.2.1) and an engine performance index model (c.f. subsection 1.2.1). The rest of the chapter provides details about the techniques required to consider inconsistent meshes, superelements and efficient iterative solvers that are required to speed-up the overall optimization elapsed time.

### 1.1.1 Elastostatics equations - strong and weak form

Let's consider an elastic body described by a 3D domain  $\Omega$  (cf. figure 1.1). One can denote its boundary,  $\partial\Omega$  and the outward normal vector  $\{\hat{n}\}$ . Furthermore  $\partial\Omega_u$  and  $\partial\Omega_\sigma$  indicate the boundary where, respectively displacements and surface traction are prescribed, so that  $\partial\Omega_\sigma \cup \partial\Omega_u = \partial\Omega$ . The hypothesis that  $\partial\Omega_u \cap \partial\Omega_\sigma = \emptyset$  was also considered.

In elastostatics problems one seeks the displacement field  $\{u\} \in H^1(\Omega)$  that solves the local

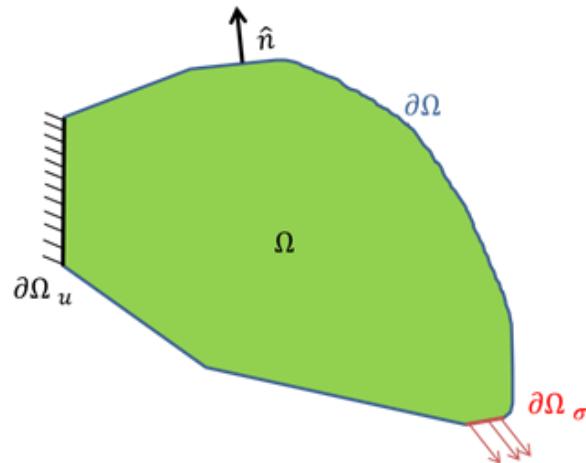


Figure 1.1: linear elastostatics problem definition

balance, boundary conditions and constitutive equations<sup>1</sup>:

$$\nabla \cdot [\sigma] + \{b\} = \{0\} \quad \forall \{x\} \in \Omega / \partial\Omega \quad (1.1)$$

$$\{u\} = \{\bar{u}\} \quad \forall \{x\} \in \partial\Omega_u \quad (1.2)$$

$$[\sigma] \cdot \{\hat{n}\} = \{\bar{t}\} \quad \forall \{x\} \in \partial\Omega_\sigma \quad (1.3)$$

$$[\sigma] = [[E]] : [\varepsilon] \quad \forall \{x\} \in \Omega \quad (1.4)$$

Where  $\{b\}$  are the body force vector,  $[\sigma]$  is the stress tensor and  $\nabla \cdot (\bullet)$  indicates the divergence operator,  $\{\bar{u}\}$  and  $\{\bar{t}\}$  are respectively the prescribed displacements field and surface force field defined on the boundary  $\partial\Omega_\sigma$  and  $\partial\Omega_u$ ,  $[[E]]$  is the fourth order elastic tensor,  $[\varepsilon] = \{[D]\} \cdot \{u\}$  is the infinitesimal strain tensor ( $\{[D]\}$  denotes the symmetric gradient operator),  $\cdot$  stands for the scalar product and  $:$  represents the double index contraction.<sup>2</sup>

In order to use a finite element approach to discretize the elastostatics equations, a weak form of equation (1.1) is introduced. By evaluating the scalar product of equation (1.1) with a compatible virtual displacement  $\{v\} \in \mathbb{V}$  and integrating it over the domain  $\Omega$ , the weak or integral formulation is obtained:

$$\int_{\Omega} \{v\} \cdot (\nabla \cdot [\sigma] + \{b\}) d\Omega = \{0\} \quad \forall \{v\} \in \mathbb{V} \quad (1.5)$$

Using the following identity, valuable for any differentiable field  $\{\alpha\}$  and for any differentiable tensor  $[\beta]$ :

$$\nabla \cdot (\{\alpha\} \cdot [\beta]) = \nabla \{\alpha\} : [\beta] + \{\alpha\} \cdot (\nabla \cdot [\beta]) \quad (1.6)$$

Equation (1.5) becomes:

$$\int_{\Omega} (\nabla \cdot (\{v\} \cdot [\sigma]) - \nabla \{v\} : [\sigma] + \{v\} \cdot \{b\}) d\Omega = \{0\} \quad \forall \{v\} \in \mathbb{V} \quad (1.7)$$

Using the divergence theorem and taking the tensor product term to the right hand side one can get:

$$\int_{\partial\Omega} (\{v\} \cdot [\sigma] \cdot \{\hat{n}\}) d\partial\Omega + \int_{\Omega} (\{v\} \cdot \{b\}) d\Omega = \int_{\Omega} (\nabla \{v\} : [\sigma]) d\Omega \quad \forall \{v\} \in \mathbb{V} \quad (1.8)$$

Finally using equations (1.2) to (1.4).

$$\int_{\partial\Omega_\sigma} \{v\} \cdot \{\bar{t}\} d\partial\Omega + \int_{\Omega} \{v\} \cdot \{b\} d\Omega = \int_{\Omega} \{[D]\} \cdot \{v\} : [[E]] : \{[D]\} \cdot \{u\} d\Omega \quad \forall \{v\} \in \mathbb{V} \quad (1.9)$$

Being both  $[\sigma]$  and  $[\varepsilon]$  symmetric, the right hand side of equation (1.9) can be simplified with

<sup>1</sup>In this context a 3D solid continuum was considered so that  $\{u\} \equiv \{u, v, w\}$ .

<sup>2</sup>In this text vectors quantity are denoted in braces, Matrix in brackets and third or fourth order arrays as Vectors of Matrices and as Matrices of Matrices, i.e. using nested braces and brackets.

the help of the following vectors:

$$\{\varepsilon\} = \begin{Bmatrix} \varepsilon_{xx} \\ \varepsilon_{yy} \\ \varepsilon_{zz} \\ \gamma_{xy} \\ \gamma_{xz} \\ \gamma_{yz} \end{Bmatrix} = \begin{Bmatrix} \frac{\partial u}{\partial x} \\ \frac{\partial v}{\partial y} \\ \frac{\partial y}{\partial z} \\ \frac{\partial w}{\partial z} \\ \frac{\partial u}{\partial y} + \frac{\partial v}{\partial x} \\ \frac{\partial u}{\partial z} + \frac{\partial w}{\partial x} \\ \frac{\partial v}{\partial z} + \frac{\partial w}{\partial y} \end{Bmatrix} = \nabla_{sym}\{u\}; \quad \{\sigma\} = \begin{Bmatrix} \sigma_{xx} \\ \sigma_{yy} \\ \sigma_{zz} \\ \tau_{xy} \\ \tau_{xz} \\ \tau_{yz} \end{Bmatrix} \quad (1.10)$$

The generalized Hook's law equation (1.4) can then be written for isotropic materials as:

$$\{\sigma\} = E \begin{bmatrix} C_1 & C_2 & C_2 & 0 & 0 & 0 \\ C_2 & C_1 & C_2 & 0 & 0 & 0 \\ C_2 & C_2 & C_1 & 0 & 0 & 0 \\ 0 & 0 & 0 & C_3 & 0 & 0 \\ 0 & 0 & 0 & 0 & C_3 & 0 \\ 0 & 0 & 0 & 0 & 0 & C_3 \end{bmatrix} \{\varepsilon\} = E[D]\{\varepsilon\} \quad (1.11)$$

Where:

$$C_1 = \frac{1-v}{(1+v)(1-2v)}; \quad C_2 = \frac{v}{(1+v)(1-2v)}; \quad C_3 = \frac{1}{2(1+v)} \quad (1.12)$$

$E$  and  $v$  are the material's Young modulus and Poisson ratio respectively. Then equation (1.9) can be written as :

$$\int_{\partial\Omega_\sigma} \{v\}^T \{\bar{t}\} d\partial\Omega + \int_{\Omega} \{v\}^T \{b\} d\Omega = \int_{\Omega} (\nabla_{sym}\{v\})^T E[D] \nabla_{sym}\{u\} d\Omega \quad \forall \{v\} \in \mathbb{V} \quad (1.13)$$

### 1.1.2 Discretization using finite element

In the standard Galerkin formulation of the finite element method the space of the solution  $\mathbb{U}$  and the space of test functions  $\mathbb{V}$  are approximated by their discrete approximation:  $\mathbb{U}^h$  and  $\mathbb{V}^h$  that are given by the linear combination of shape functions defined in each element<sup>3</sup>. so that the test function and the displacement field can be approximated as follows:

$$\{v\} = [N]^{(h)} \{v\}^{(h)} \quad \{u\} = [N]^{(h)} \{u\}^{(h)} \quad (1.14)$$

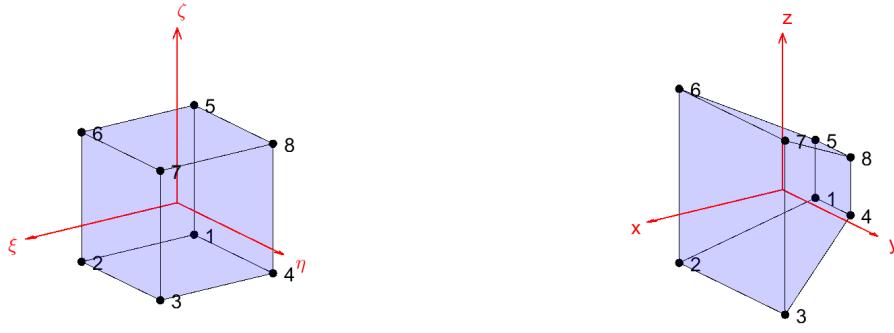
In which  $\{v\}^{(h)}$  and  $\{u\}^{(h)}$  are the vectors containing the corresponding field values at each node for each degree of freedom,  $[N]^{(h)}$  is the  $N_d \times N_d N$  matrix containing,  $N_d$  is the number of DOFs per node,  $N$  is the number of nodes of the discretization. Accordingly in a 3-D mesh with solid elements  $[N]^{(h)}$  has the following structure:

---

<sup>3</sup>The standard Galerkin approximation is a well-established approach that produces a symmetric stiffness matrix consistent with the classic variational formulation. More generally, the spaces of functions of  $\mathbb{U}^h$  and  $\mathbb{V}^h$  may not be the same, as it is the case for the Petrov-Galerkin methods.

$$[N]^{(h)} = \begin{bmatrix} N_1 [I]_d & N_2 [I]_d & \cdots & N_j [I]_d & \cdots & N_N [I]_d \end{bmatrix} \quad (1.15)$$

Where  $[I]_d$  is the  $N_d \times N_d$  identity matrix and  $N_j$  are the shape functions associated with the  $j^{th}$  node. These functions are often chosen to be Lagrange basis polynomials defined on each finite element. For ease of implementation, FEA software makes the use of reference elements (c.f. figure 1.2a) that renders the definition and the integration of shapes functions easier. The mapping between reference and actual geometry within a finite element is also made by



(a) Reference element in the  $(\xi, \eta, \zeta)$  space   (b) Corresponding mapped element in the  $(x, y, z)$  space

Figure 1.2: Classic Mapping in FEA

the same shape functions as follows:

$$\{X_g\} = \begin{Bmatrix} x \\ y \\ z \end{Bmatrix} = \begin{bmatrix} \{x_n\}^T \\ \{y_n\}^T \\ \{z_n\}^T \end{bmatrix} \{N_n(\xi, \eta, \zeta)\} \quad (1.16)$$

Where  $\{x_n\}, \{y_n\}, \{z_n\}$  are the column vectors containing the  $x, y$  and  $z$  coordinates of element nodes, and  $\{N_n(\xi, \eta, \zeta)\}$  is the column vector containing the value of shape functions relative to element nodes. To determine the form of the shape functions one can observe that putting in input the element node reference coordinates  $\xi_i, \eta_i, \zeta_i$ , for  $i = 1, 2, \dots, N$ , back into equation (1.16) one should find the corresponding nodal coordinates  $x_i, y_i, z_i$ . This means that all shape functions have to satisfy  $N$  different interpolation conditions. For this reason, in the case of 8 node hexahedral elements, each shape function is chosen to have the following

form:

$$N_i(\xi, \eta, \zeta) = \left\{ 1 \quad \xi \quad \eta \quad \zeta \quad \xi\eta \quad \xi\zeta \quad \eta\zeta \quad \xi\eta\zeta \right\} \begin{Bmatrix} a_{i000} \\ a_{i100} \\ a_{i010} \\ a_{i001} \\ a_{i110} \\ a_{i101} \\ a_{i011} \\ a_{i111} \end{Bmatrix} = \{p(\xi, \eta, \zeta)\}^T \{a_i\} \quad (1.17)$$

then one can also write:

$$\{N_n(\xi, \eta, \zeta)\} = \begin{bmatrix} \{a_1\}^T \\ \{a_2\}^T \\ \vdots \\ \{a_8\}^T \end{bmatrix} \{p(\xi, \eta, \zeta)\} = [A] \{p(\xi, \eta, \zeta)\} \quad (1.18)$$

To find the 64 coefficients of  $[A]$  one can employ the 8 interpolation conditions specific to each shape function:

$$\begin{bmatrix} \{x_n\}^T \\ \{y_n\}^T \\ \{z_n\}^T \end{bmatrix} = \begin{bmatrix} \{x_n\}^T \\ \{y_n\}^T \\ \{z_n\}^T \end{bmatrix} [A] \begin{bmatrix} \{p(\xi_1, \eta_1, \zeta_1)\} & \{p(\xi_2, \eta_2, \zeta_2)\} & \cdots & \{p(\xi_8, \eta_8, \zeta_8)\} \end{bmatrix} \quad (1.19)$$

Where  $\xi_i, \eta_i, \zeta_i$  are nodal reference coordinates and can be obtained as the components of the following vectors:

$$\{\xi\} = \begin{Bmatrix} -1 \\ 1 \\ 1 \\ -1 \\ -1 \\ 1 \\ 1 \\ -1 \end{Bmatrix} \quad \{\eta\} = \begin{Bmatrix} -1 \\ -1 \\ 1 \\ -1 \\ -1 \\ 1 \\ 1 \\ 1 \end{Bmatrix} \quad \{\zeta\} = \begin{Bmatrix} -1 \\ -1 \\ -1 \\ -1 \\ 1 \\ 1 \\ 1 \\ 1 \end{Bmatrix} \quad (1.20)$$

To satisfy equation (1.19) one must have:

$$[A] = \left[ \begin{array}{cccc} \{p(\xi_1, \eta_1, \zeta_1)\} & \{p(\xi_2, \eta_2, \zeta_2)\} & \cdots & \{p(\xi_8, \eta_8, \zeta_8)\} \end{array} \right]^{-1} \quad (1.21)$$

Using the discretization of equation (1.14), the problem of equation (1.9) can be written as the well-known system of equation:

$$[K]^{(h)} \{u\}^{(h)} = \{f\}^{(h)} \quad (1.22)$$

Typically  $[K]^{(h)}$  and  $\{f\}^{(h)}$  are called *stiffness matrix* and *load vector*. They are defined as:

$$K_{i,j}^{(h)} = \int_{\Omega} \left( [\nabla_{sym} N_i]^{(h)} \right)^T E [D] [\nabla_{sym} N_j]^{(h)} d\Omega \quad (1.23)$$

$$f_i^{(h)} = \int_{\partial\Omega_{\sigma}} ([N_i]^{(h)} \cdot \{\bar{t}\}) d\partial\Omega + \int_{\Omega} ([N_i]^{(h)} \cdot \{b\}) d\Omega \quad (1.24)$$

Here the column vectors  $[N_i]^{(h)}$  and  $[N_j]^{(h)}$  are respectively the i-th and j-th column of  $[N]^{(h)}$ . The integrals of equations (1.23) and (1.24) can be decomposed as the sum of the integrals computed on each finite element:

$$K_{i,j}^{(h)} = \sum_{el=1}^{n_{el}} E_{el} \int_{\Omega_{el}} \left( [\nabla_{sym} N_i]^{(h)} \right)^T [D] [\nabla_{sym} N_j]^{(h)} d\Omega_{el} \quad (1.25)$$

$$f_i^{(h)} = \sum_{el=1}^{n_{el}} \int_{\partial\Omega_{\sigma} \cap \partial\Omega_{el}} ([N_i]^{(h)} \cdot \{\bar{t}\}) d\partial\Omega_{el} + \int_{\Omega_{el}} ([N_i]^{(h)} \cdot \{b\}) d\Omega_{el} \quad (1.26)$$

Where the Young's modulus was supposed to be uniform over each finite element. These integrals are usually evaluated by Gauss quadrature over each element. The mapping of equation (1.16) can then be employed:

$$K_{i,j}^{(h)} = \sum_{el=1}^{n_{el}} E_{el} \sum_{k=1}^{N_{GP}} \left( [\nabla_{sym} N_{ik}]^{(h)} \right)^T [D] [\nabla_{sym} N_{jk}]^{(h)} |[J_k]| \omega_k \quad (1.27)$$

$$f_i^{(h)} = \sum_{el=1}^{n_{el}} \sum_{l=1}^{N_{GP}} ([N_{il}]^{(h)} \cdot \{\bar{t}_l\}) |[J_l]| \omega_l + \sum_{k=1}^{N_{GP}} ([N_{ik}]^{(h)} \cdot \{b_k\}) |[J_k]| \omega_k \quad (1.28)$$

Where the notation  $[\nabla_{sym} N_{ik}]$  means that the i-th column of the matrix containing  $[\nabla_{sym} N_i]$  is computed on the k-th Gauss point of the el-th element,  $\omega_k$  refers to the integration weights of the Gauss-Legendre integration rule chosen to compute each integral. Moreover due to the variable change, the Jacobian determinant of the geometrical mapping  $|[J_k]|$  has to be computed in each Gauss point of each element in the FEM. Jacobian determinant can be computed as:

$$|[J]| = \left| \left[ \frac{\partial \{X_g\}}{\partial \xi}, \frac{\partial \{X_g\}}{\partial \eta}, \frac{\partial \{X_g\}}{\partial \zeta} \right] \right| = \left| \begin{bmatrix} \{x_n\}^T \\ \{y_n\}^T \\ \{z_n\}^T \end{bmatrix} [A] \begin{bmatrix} \frac{\partial \{p\}}{\partial \xi}, \frac{\partial \{p\}}{\partial \eta}, \frac{\partial \{p\}}{\partial \zeta} \end{bmatrix} \right| \quad (1.29)$$

In this expression one can observe that both  $[A]$ , and  $\left[ \frac{\partial \{p\}}{\partial \xi}, \frac{\partial \{p\}}{\partial \eta}, \frac{\partial \{p\}}{\partial \zeta} \right]$  only depend on the reference element nodal coordinates and on the Gauss point position in the reference space and not on the nodal coordinates of the finite element. For the expression of  $[\nabla_{sym} N_i]^{(h)}$  one

can apply the definition of  $\nabla_{sym}$  operator in equation (1.9) to the i-th column of  $[N]^{(h)}$ :

$$[\nabla_{sym} N_i]^{(h)} = \left\{ \begin{array}{l} \frac{\partial N_{1i}}{\partial x} \\ \frac{\partial N_{2i}}{\partial y} \\ \frac{\partial N_{3i}}{\partial z} \\ \frac{\partial N_{1i}}{\partial y} + \frac{\partial N_{2i}}{\partial x} \\ \frac{\partial N_{1i}}{\partial z} + \frac{\partial N_{3i}}{\partial x} \\ \frac{\partial N_{2i}}{\partial z} + \frac{\partial N_{3i}}{\partial y} \end{array} \right\} \quad (1.30)$$

Where  $N_{1i}, N_{2i}, N_{3i}$  indicate the first, the second and the third component of the i-th column of  $[N]^{(h)}$ . Computing  $[\nabla_{sym} N_i]^{(h)}$  is a non-trivial operation since shape functions are expressed in terms of reference variables  $(\xi, \eta, \zeta)$ , that map through equation (1.16) the  $(x, y, z)$  space. So the inverse mapping between reference and geometric space is an implicit set of functions that often cannot be reversed explicitly. To compute the shape function derivatives with respect to space variable one can use Jacobian matrix definition <sup>4</sup>:

$$\left[ \frac{\partial N_{li}}{\partial x}, \frac{\partial N_{li}}{\partial y}, \frac{\partial N_{li}}{\partial z} \right] = [J]^{-1} \left\{ \begin{array}{l} \frac{\partial N_{li}}{\partial \xi} \\ \frac{\partial N_{li}}{\partial \eta} \\ \frac{\partial N_{li}}{\partial \zeta} \end{array} \right\} \quad (1.31)$$

Thanks to these relationships, the system of balance equation can be expressed in terms of a linear system of equations. This system is actually singular, in fact as a last step one needs to apply the boundary conditions (equation (1.2)). For brevity one will here after refer to stiffness matrix as  $[\mathbf{K}_{tot}]$ , to the load vector as  $\{f_{tot}\}$  and to the displacement vector as  $\{u\}$ :

$$[\mathbf{K}_{tot}] \{u_{tot}\} = \{f_{tot}\} \quad (1.32)$$

The load vector can actually be decomposed in two components: the external load vector  $\{F\}$  coming from applied load and the reaction forces vector  $\{R\}$  coming from the boundary conditions so that:

$$\{f_{tot}\} = \{F\} + \{R\} \quad (1.33)$$

To apply the boundary conditions it is useful to adopt a partition of degrees of freedoms in two sets: the set of constrained DOFs  $\{u_b\}$  and the set of free DOFs  $\{u_f\}$ . According to this partition equation (1.22) becomes:

$$[\mathbf{K}_{bb}] \{u_b\} + [\mathbf{K}_{bf}] \{u_f\} = \{R_b\} + \{F_b\} \quad (1.34)$$

$$[\mathbf{K}_{fb}] \{u_b\} + [\mathbf{K}_{ff}] \{u_f\} = \{R_f\} + \{F_f\} \quad (1.35)$$

The reaction force is applied only on constrained DOFs so that  $\{R_f\} = \{0\}$ . If all rigid modes are constrained by the boundary conditions, then  $[\mathbf{K}_{ff}]$  is non-singular. In this case,

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<sup>4</sup> Jacobian matrix of the geometric mapping is nonsingular when finite elements are not twisted and have all distinct nodes

the system of equations (1.34)-(1.35) can be solved with respect to the unknowns  $\{u_f\}, \{R_b\}$ :

$$\{u_f\} = [\mathbf{K}_{ff}]^{-1} (-[\mathbf{K}_{fb}] \{u_b\} + \{F_f\}) \quad (1.36)$$

$$\{R_b\} = [\mathbf{K}_{bb}] \{u_b\} + [\mathbf{K}_{bf}] \{u_f\} - \{F_b\} \quad (1.37)$$

Another way of dealing with boundary conditions that can be more suitable for Multi-Grid solver is replacing the rows relative to constrained DOFs directly with the corresponding kinematic constraint equations. To keep stiffness matrix symmetry for non-homogeneous Boundary Conditions, the term  $\mathbf{K}_{fb} \{u_b\}$  has to be considered directly in the right hand side.

$$[\mathbf{A}_{bb}] \{u_b\} = \{b_b\} \quad (1.38)$$

$$[\mathbf{K}_{ff}] \{u_f\} = \{F_f\} - [\mathbf{K}_{fb}] \{u_b\} \quad (1.39)$$

Note that for simple clamping conditions  $[\mathbf{A}_{bb}]$  is the identity matrix and  $\{b_b\} = 0$  so that also  $[\mathbf{K}_{fb}] \{u_b\}$  is zero. The advantage of this alternative formulation is that the kinematic constraints are directly included in the global stiffness matrix, making it easier to make the transfer among discretization levels for multigrid solvers, as it will be discussed later in this chapter. In most of commercial software a post processing can be applied to compute both the deformation field and the stress field on each finite element of the FEM. To do that one needs to apply equation (1.9) and (1.14). As the shape functions are  $C^0$  function, their derivatives are actually discontinuous passing from a finite element to the other. Moreover it has been shown that stress and deformations computed through the FEA are the most accurate when computed at the Gauss Points [Zlámal, 1978, Zhang and Naga, 2006]. For this reason in the preprocessing phase, when the stiffness matrix and the load vector are assembled, the following matrixes are commonly stored to be reused in the post processing phase:

$$\{\varepsilon\} = [B_{gp}] \{u_{el}\} \quad \{\sigma\} = [D] [B_{gp}] \{u_{el}\} \quad (1.40)$$

Where  $[B_{gp}]$  is a  $6 \times 24$  matrix called stiffness-displacement as it links the value of deformations in a given Gauss point with  $\{u_{el}\}$  that indicates the  $24 \times 1$  vector of DOFs for the corresponding element.

## 1.2 A generic engine FEM for fuel consumption variation computation

As discussed in the introduction, to compute the effect of a given design structure on tip clearance variation control, one needs to consider a mechanical model of the engine, nacelle and pylon (that hereafter can also be described as Power-Plant System (PPS)). This requires to build the model of each component and then to assemble them together. In this section the engine finite element model employed as demonstrator is introduced. Then the post processing steps needed for the computation of a global performance index with respect to tip clearance variation is introduced. This will be described as thrust specific fuel consumption variation due to its link to the actual engine performance deterioration, induced by clearance variation.

The reader can also observe that for the reasons discussed in the introduction, this parameter is also correlated to other issues such as engine life and operability.

### 1.2.1 Structural Model

The mechanical behavior of both engine and PPS structure is usually studied using a Finite Element Model, integrating complex load combinations. Here a generic engine model, that will be used throughout this work, is introduced (c.f. figure 1.3). This model is a simplification

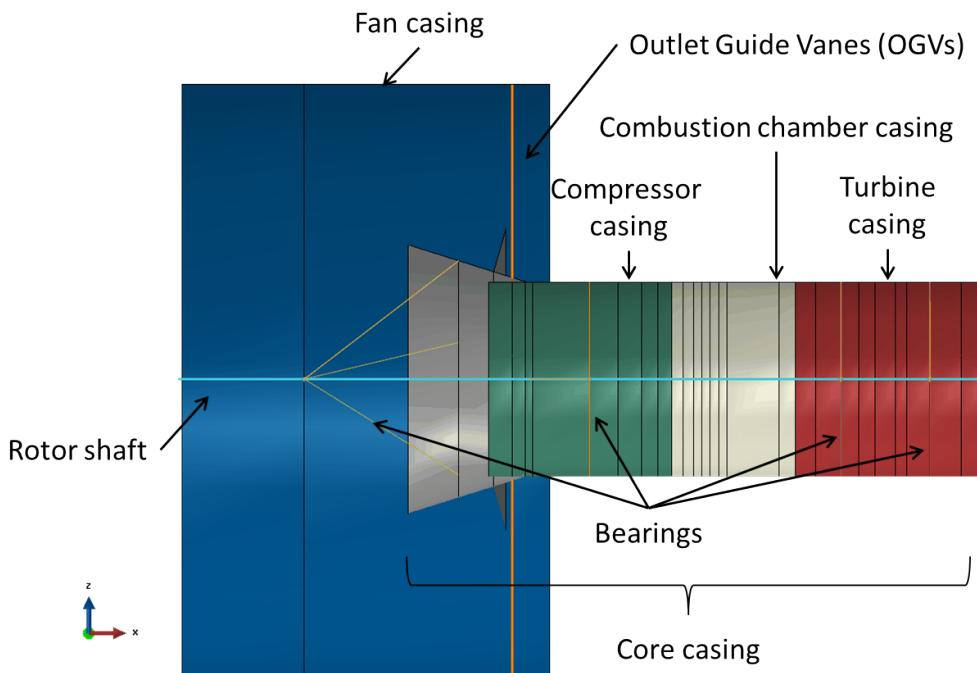


Figure 1.3: Proposed Whole Engine Model (WEM). Colors identify zone with different properties. The connection between rotor and stator is represented by rigid connections at bearing locations.

of an industrial engine model that in general can have a fan casing structure, a core casing structure and multiple shafts for rotors. The following assumptions are made regarding the model's mechanical behavior. Rotor shafts are modeled using just one beam model. Both fan casing and core casing are represented using shell models. Different properties are employed for each engine module casing. The connection between rotor and stator is made through four rigid connections that represent stator vane and bearing housing in the engine. Outlet guide vanes which connect the core casing to the fan casing are represented with radial beam model tied to each interface of the engine. The thermal and centrifugal expansion of rotor blades are not considered in this analysis and nonlinearities are neglected. Only aerodynamic loads linked with the aircraft maneuvers are considered for simplicity. These loads are applied statically. A general 3D solid design zone is integrated between the wing and the engine (see gray elements in figure 1.4). This is the zone where the topology optimization will need to find

the optimal material placement. For simplicity and since this does not have a significant effect on optimization, the air inlet and the nacelle structure were not considered. The engine core casing and the design zone meshes are tied at specific region of compressor and turbine casing. For this reason the solution of the topology optimization problem will be connected to the engine at these same regions. This will avoid solutions connected on prohibited region such as the combustion chamber casing. The techniques employed for the connection between engine casing and design zone mesh will be detailed in section 1.3. Eight node brick elements were

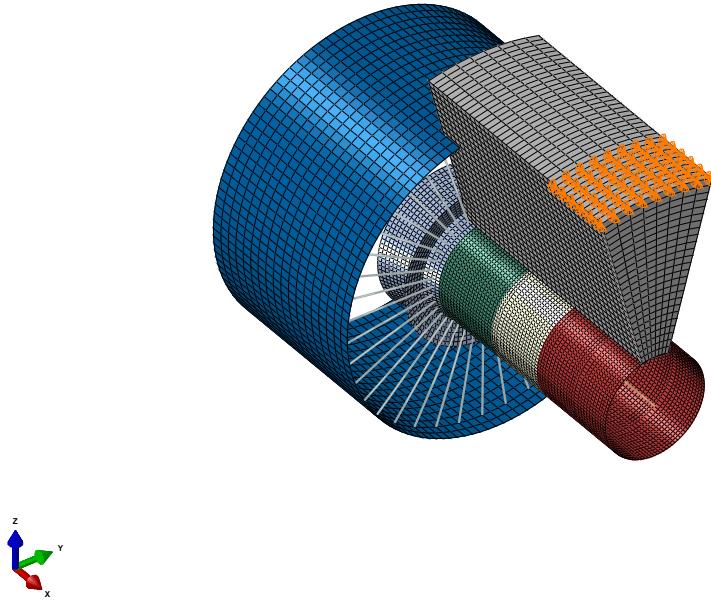


Figure 1.4: Engine and design zone (in gray) finite element model. The engine model is made of 9976 finite elements, 9312 linear quadrilateral elements of type and 664 beam elements. The solid design zone is clamped at the orange points representing a clamping to the wing.

used to mesh the design zone and full eight points integration was employed for elementary stiffness matrix and stress evaluation. The design zone model is clamped at the wing interface position. The axial load on the engine is applied with concentrated forces on the shaft nodes and with distributed loads on the engine casing. Vertical and lateral loads are distributed uniformly on the circumference of fan stage. The commercial software Abaqus 13.2 was employed for mesh generation and load case application.

### 1.2.2 Performance Model

The performance function that is considered in this work is a simplification of the real Thrust Specific Fuel Consumption (TSFC) variation induced by the considered loading conditions.

Engine fuel consumption could be impacted in different ways by the pylon and casing design. Here the only effect of closures variations is taken into account. The tip clearance variation is described in figure 1.5, and defined in equation (1.41).

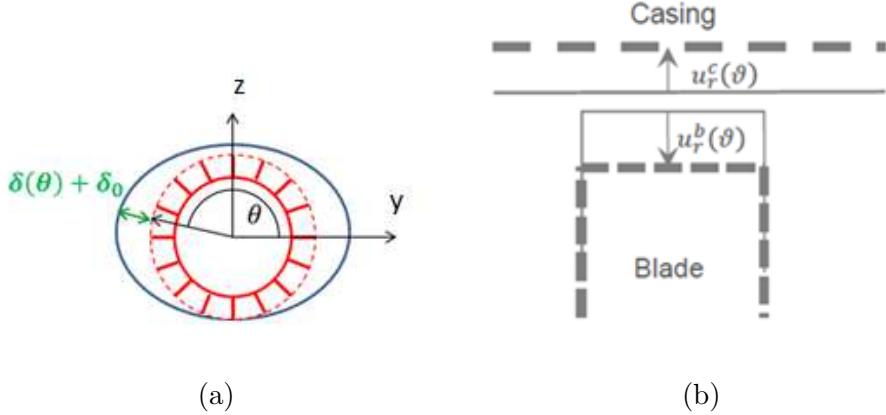


Figure 1.5: Rotor and stator displacements under maneuvers loads: (a)  $z - y$  engine section diagram for a given rotor stage. The red structure stands for the rotor, the blue curve for the deformed stator. (b) diagram illustrating radial displacement effect. The resulting tip clearance is the sum of the initial clearance  $\delta_0$  and its variation induced by the engine deformation  $\delta(\theta)$

$$\delta(\theta) = u_r^c(\theta) - u_r^b(\theta) = \{r(\theta)\}^T \cdot (\{u^c(\theta)\} - \{u^b(\theta)\}) \quad (1.41)$$

Where  $u_r^c(\theta)$  and  $u_r^b(\theta)$  are the radial displacements of casing and rotor blade tip respectively at the angular position  $\theta$  and  $\{r(\theta)\}$  is the radial unit vector at the angle  $\theta$ . Considering an angular discretization of the circumference given by the stator mesh in the  $s^{th}$  engine stage,  $\theta_{(s)}^i$  the tip clearance variation at the angular section  $i$  is then:

$$\{\delta_{(s)}\}_i = \{r(\theta_{(s)}^i)\}^T \cdot (\{u^c(\theta_{(s)}^i)\} - \{u^b(\theta_{(s)}^i)\}) \quad (1.42)$$

That is actually a system of linear relationships that can be expressed in the matrix form:

$$\{\delta_{(s)}\} = [\gamma_{(s)}] \{U\} \quad (1.43)$$

where  $\{U\}$  is the displacements vector. To characterize a stage's contribution to the overall engine performance, tip-clearance's variation Root Mean Square (RMS) is evaluated at each stage:

$$R_{(s)} = \sqrt{\frac{\{\delta_{(s)}\}^T \{\delta_{(s)}\}}{N_{(s)}}} \quad (1.44)$$

where  $N_{(s)}$  is the number of nodes used for the angular discretization of the  $s^{th}$  stage. In this thesis, a simple model to link the tip clearances to the TSFC performance is used. The variation in TSFC due to aircraft maneuvers is considered here as a linear combination of the RMS values per stage:

$$\Delta TSFC\% = \sum_{s=1}^{ns} \frac{R_s}{l_{(s)}} 100 \quad (1.45)$$

where  $ns$  is the number of stages of the engine and  $l_{(s)}$  the stage blade height. When considering several maneuvers, in order to get a single value of performance one can adopt several approaches. In an industrial modeling attempt, the TSFC variation relative to a given load case needs to be weighted using the probability of occurrence of the corresponding flight condition in the product life. In this attempt one should consider:

$$\Delta TSFC\% = \sum_{lc=1}^{N_l} f_{lc} (\Delta TSFC\%)_{lc} \quad (1.46)$$

Where  $N_l$  is the number of flight points considered,  $(\Delta TSFC\%)_{lc}$  and  $f_{lc}$  is the corresponding TSFC variation and the probability of occurrence respectively. To give an insight on this formula, one can observe that for each aircraft family one will have different weighting for each loading condition. In reality one can observe that this model does not reflect the actual design process of an actual engine, neither the correlation between several operating conditions. A turbofan engine initial clearance  $\delta_0$  should in fact be designed to avoid casing to blade contact in any operating conditions. This means that:

$$(\delta_0)_s = \max \left( 0, \min_l \left( (\delta_{(s)})_l \right) \right) \quad (1.47)$$

By the use of equation 1.46, this time one can observe that the effect of load cases generating larger negative tip clearance variations, will also affect the performance in situations where the tip clearance variations are smaller. This means that load cases having larger negative variations will be the one that affect the most the specific fuel consumption variation. Due to that observation when considering several loading conditions a good indicator of the final performance can be associated to the highest TSFC variation (which is considered to be also linked to load cases that generate the largest negative tip clearance variations):

$$\Delta TSFC\% = \max_l (\Delta TSFC\%)_l \quad (1.48)$$

### 1.3 Tying inconsistent meshes

An important issue that needs to be addressed is the mesh inconsistency between engine and design zone. In fact the Engine Finite Element Model in the current practice is built by the engine manufacturer and is not initially intended for use in a topology optimization framework. For this reason in order to freely mesh the design zone, the connection between these two non-overlapping domains need to be established through their interface DOFs.

The first hypothesis made is that only translational DOFs of engine shell elements have to be

considered for kinematic tying. For this reason the engine condensation (c.f. subsection 1.4.1) was performed around the retained nodes translational DOFs and not for rotational DOFs, that are therefore not constrained. The external skin of the engine's shell elements need to have the same displacements as the external surface of the solid elements. Many techniques can be employed for dealing with this problem as described in the review [Coniglio et al., 2018a]. Situations arise where a finite element model needs to be assembled from parts that can present non-matching grids (meshes). In these cases the displacement at the interface is not uniquely determined and special techniques have to be used to take into account the non-conforming interface [McGee and Seshaiyer, 2005].

This can be the case:

- When two parts have to be meshed independently for practical reasons [Barlow, 1982, Quiroz and Beckers, 1995]. This can be due to the impossibility to get good quality meshes for two adjacent components or due to models being generated independently by different teams or for different purposes, or even for modularity if one component model has to be integrated in different product models.
- When different physical phenomena are studied with different discretizations: for example in acoustics [Flemisch et al., 2012]. One of the most popular applications is Fluid-Structure Interaction (FSI) [De Boer et al., 2007, de Boer et al., 2008, Hou et al., 2012, Beckert and Wendland, 2001, Cebral and LČohnerē, 1997, Löhner et al., 1998, Farhat et al., 1998]. The mesh needed for the fluid is much finer than the one that needs to be used for solid. Important computation time can be saved using mesh projection techniques to connect different meshes.
- When the domain decomposition is required ([Brandt, 2002, Feyel and Chaboche, 2000, Duval et al., 2016, Farhat and Roux, 1991, Mandel, 1993]). The communication between the large scale and lower scale meshes is often done with a mesh projection operator. Such domain decomposition is for example useful when a much finer mesh is needed in a localized region of a structure (e.g. in crack propagation analysis [Duval et al., 2016, Lloberas-Valls et al., 2012b, Lloberas-Valls et al., 2012a]).
- When contact between meshes is activated in the simulation [Wriggers, 1995, Shillor et al., 2004, Sofonea et al., 2005, Christensen et al., 1998]. In the most general situation rebounding and sliding can appear the exchange of information between meshes changes with the grid configurations. In this application the mesh projection operator has to be computed fast enough so that its evaluation can be repeated in the analysis.

### 1.3.1 State of the art

The main challenge of the aforementioned situation is to ensure the continuity of the displacement field at the non-conforming interface. In the general case of non-matching interfaces, the displacement field will be continuous at the cost of an over-constraint of the interface [Rixen, 1997]. To avoid this phenomenon, typically both strong and weak coupling can be employed to satisfy the compatibility of the solution at the interface node DOFs.

### 1.3.1.1 Strong and weak formulations

The strong coupling techniques are also referred to as collocation techniques [Rixen, 1997, Aminpour et al., 1995] or node to segment [Wriggers, 1995], as one constraint equation is assigned to each interface node DOF. In the weak coupling, also called segment to segment [Wriggers, 1995] methods, on the other hand, the continuity is written in an integral or averaged sense. To avoid interface stiffening, the weak formulation is preferred [Aminpour et al., 1995, Fish et al., 1995]. For both these "classic" approaches, one surface is chosen to be the one that will produce the constraint equations (slave surface) and the other is considered as interpolating (master surface). Each DOF of the slave surface is then written as a linear combination of the DOFs of the master surface. Once that this linear system of constraint equations is written, the final displacement solution can be obtained by the elimination method [Barlow, 1982] i.e. substituting the slave node DOFs variable with their expression in terms of master node DOFs. Since the solution of a static solid analysis can also be considered as a minimization problem, the linear constraints can be imposed using the Lagrange multipliers approach [Bertsekas, 2014, Babuška, 1973]. A variable is therefore created for each constraint that has to be satisfied and used to build the Lagrangian function. The stationarity conditions of the Lagrangian function give the linear system of equations used to find the nodal displacement and the interface Lagrange multiplier variables. The latter also have the physical interpretation of interface internal forces. One of the most used techniques that utilizes Lagrange multipliers in a continuous form is the Mortar approach [Bernardi et al., 1993, Bernardi, 1989, Puso, 2004, Puso and Laursen, 2004]. The distribution functions for the Lagrange multipliers and shape functions for the finite elements should be properly selected to fulfill the Ladyzhenskaya-Babuška-Brezzi (LBB) condition (also known as the inf-sup condition) [Babuška and Narasimhan, 1997] in order to guarantee that both discretizations converge to the correct solution with the mesh refinement.

### 1.3.1.2 Alternative and improved approaches

Other numerical schemes followed, inspired by the optimization community. In the PhD dissertation of Rixen [Rixen, 1997] one can find a method based on the augmented Lagrangian method in optimization. In the review of Barlow [Barlow, 1982] another method based on the external penalty approach is presented. The main difficulty of this approach consists in the choice of the penalty factor. Deparis et al. introduced in [Deparis et al., 2016] a new approach based on the simultaneous continuity of the displacement and the internal loads per unit of area at the interface. This promising technique has been tested in the case of a simple patch test and more complex fluid structure interaction problems. In those cases the method reveals the same performance as Mortar, with a much smaller computational effort and programming complexity. In order to pass *a priori* the simple stress patch test even for curved interfaces where Mortar typically fails, Park et al [Park et al., 2002] developed a method based on the introduction of a third displacement field whose nodes are placed based on an equilibrium equation of moments. The resulting scheme is however quite complex and the applications considered in [Park et al., 2002] were only for a simple 2D case and a planar interface on a

3D case. For overlapped domains with non-consistent meshes the Arlerquin method has also been proposed [Dhia and Rateau, 2005, Dhia et al., 2008]. More invasive approaches were developed by Cho et al. [Cho et al., 2005, Cho and Im, 2006a, Cho and Im, 2006b], Lim et al. [Lim et al., 2007a, Lim et al., 2007b], Kim et al. [Kim, 2003] and Duarte [Duarte and Kim, 2008], where the idea is to modify the Element shape functions adding some nodes to the element of the interface or enriching the shape functions. In Dohrman et al. [Dohrmann et al., 2000a, Dohrmann et al., 2000b] the slave element locations and formulations are modified in order to transform a non-conforming interface into a conforming one. In Tian et al. [Tian and Yagawa, 2007] interface elements formulation is replaced with a meshless formulation. In this way the coupled analysis becomes straightforward but the implementation of a special formulation for interface elements is not as simple as the application of multi-point constraints (MPCs) at the interface. A Least Square Method can be found in Bochev et al. [Bochev and Day, 2007] in order to pass a simple patch test. This method requires nevertheless the meshes to be perturbed at the interface to avoid gaps between curved interfaces, which render the approach more complex to implement. Bitencourt et al. [Bitencourt et al., 2015] introduced a new method that assembles Coupling Finite Elements (CFEs) at the interface to build the constraints equation between interfaces Degrees of Freedom (DOFs). This approach was studied for 3D planar, 2D planar or 2D curved interfaces. A recent method was developed by Cafiero et al. [Cafiero et al., 2016] on the base of the previous developments of Nitsche [Nitsche, 1971], Becker [Becker et al., 2003], Heintz [Heintz and Hansbo, 2006], Olivier [Oliver et al., 2009] and Hartman [Hartmann et al., 2009]. This approach, inspired from the contact community avoids the expensive segment to segment projections needed for the mortar approach introducing a particular formulation in the gap between the non-conforming meshes at the interface. The Domain Interface Method (DIM) [Cafiero et al., 2016] was also extended to the case of mixed fields as can be encountered in multiphysics problems [Lloberas-Valls et al., 2017]. Another similar approach introduced virtual gap elements at the interface of the domains and imposes a zero strain condition to these elements [Song et al., 2017].

### 1.3.2 Domain partition

The same problem is considered here but this time the domain  $\Omega$  is partitioned into two subdomains  $\Omega_1$  and  $\Omega_2$  (cf. figure 1.6). The general case with  $m$  subdomain can be easily derived from this case. The surface used for the partition is unique ( $\Gamma_1 \equiv \Gamma_2 \equiv \Gamma$ ), as a consequence:

$$\{u\}_{\Gamma_1} = \{u\}_{\Gamma_2} \quad \forall \{x\} \in \Gamma \quad (1.49)$$

$$[\sigma]_{\Gamma_1} \cdot \{\hat{n}\}_{\Gamma_1} = \{t\}_{12} = -\{t\}_{21} = -[\sigma]_{\Gamma_2} \cdot \{\hat{n}\}_{\Gamma_2} \quad \forall \{x\} \in \Gamma \quad (1.50)$$

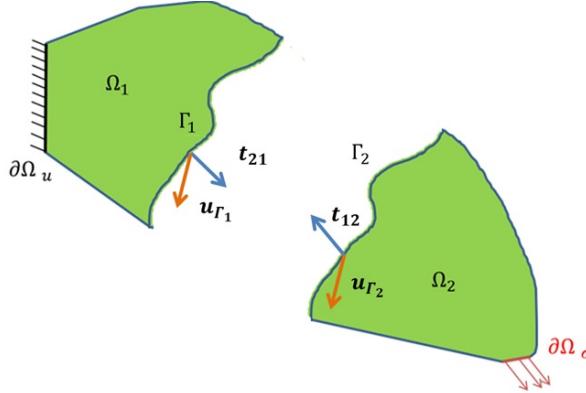


Figure 1.6: Partition of  $\Omega$  in two subdomains  $\Omega_1, \Omega_2$

Moreover taking in account the partition, it is possible to show that:<sup>5</sup>

$$\int_{\Gamma_1} (\{v\} \cdot \{t\}_{21}) d\Gamma_1 + \int_{\partial\Omega_{\sigma_1}} (\{v\} \cdot \{\bar{t}\}) d\partial\Omega + \int_{\Omega_1} (\{v\} \cdot \{b\}) d\Omega = \int_{\Omega_1} (\{[D]\} \{v\} : [[E]] : \{[D]\} \{u\}) d\Omega \quad (1.51)$$

$$\int_{\Gamma_2} (\{v\} \cdot \{t\}_{12}) d\Gamma_2 + \int_{\partial\Omega_{\sigma_2}} (\{v\} \cdot \{\bar{t}\}) d\partial\Omega + \int_{\Omega_2} (\{v\} \cdot \{b\}) d\Omega = \int_{\Omega_2} (\{[D]\} \{v\} : [[E]] : \{[D]\} \{u\}) d\Omega \quad (1.52)$$

That may be combined with equation (1.50) to get equation (1.13). As expected the virtual partition into subdomain is a choice that does not affect the solution of the elastostatics problem.

### 1.3.3 Domain decomposition with consistent discretization

When a finite element discretization is used to solve the elastostatic problems over the subdomains  $\Omega_1$  and  $\Omega_2$ , the interface will be discretized in the surfaces  $\Gamma_1^h$  and  $\Gamma_2^h$ . It is possible to refer to consistent discretization in the case where the nodes of the interface are on the same positions for both subdomains  $\Omega_1$  and  $\Omega_2$  and the shape functions of each element on both sides of the interfaces are the same.

In this case the domain decomposition is not a concrete issue. In fact the communication between the degrees of freedom (DOFs) of the two discretizations is straightforward. Nevertheless one introduce a partition of the DOFs that will be used for each of the latter approaches discussed in this work. It is possible to define  $\{u\}_1^{(h)}, \{u\}_2^{(h)}, \{u\}_{\Gamma_1}^{(h)}, \{u\}_{\Gamma_2}^{(h)}$  that denote re-

<sup>5</sup>In these expression the terms  $\int_{\Gamma_1} (\{v\} \cdot \{t\}_{21}) d\Gamma_1$  and  $\int_{\Gamma_2} (\{v\} \cdot \{t\}_{12}) d\Gamma_2$  are due to the work of internal forces at the boundary. In a variational approach it would naturally vanish considering the work over the whole structure.

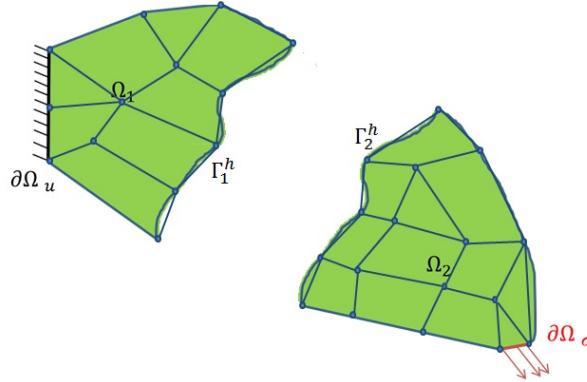


Figure 1.7: Consistent discretization of the partitioned domain

spectively the DOFs of nodes inside  $\Omega_1$  but not on the interface  $\Gamma_1$ , the DOFs of nodes inside  $\Omega_2$  but not on the interface  $\Gamma_2$ , the DOFs of the nodes of the interface  $\Gamma_1$  and the DOFs of the nodes on the interface  $\Gamma_2$  respectively. The discretized form of equations (1.51)-(1.52) is then:

$$\begin{bmatrix} [\mathbf{K}]_{1,1} & [\mathbf{K}]_{1,\Gamma_1} & [0] & [0] \\ [\mathbf{K}]_{\Gamma_1,1} & [\mathbf{K}]_{\Gamma_1,\Gamma_1} & [0] & [0] \\ [0] & [0] & [\mathbf{K}]_{\Gamma_2,\Gamma_2} & [\mathbf{K}]_{\Gamma_2,2} \\ [0] & [0] & [\mathbf{K}]_{2,\Gamma_2} & [\mathbf{K}]_{2,2} \end{bmatrix} \begin{Bmatrix} \{u\}_1 \\ \{u\}_{\Gamma_1} \\ \{u\}_{\Gamma_2} \\ \{u\}_2 \end{Bmatrix} = \begin{Bmatrix} \{f\}_1 \\ \{f\}_{\Gamma_1} + \{r\}_{\Gamma_1} \\ \{f\}_{\Gamma_2} + \{r\}_{\Gamma_2} \\ \{f\}_2 \end{Bmatrix} \quad (1.53)$$

The residuals  $\{r\}_{\Gamma_1}$  and  $\{r\}_{\Gamma_2}$  are the expression of the internal forces acting on the interface DOFs of each surface. Equations (1.49) and (1.50) are easily interpreted in this case, since the node in the same position belonging to both meshes will merge and have the same DOFs. This can be expressed as:

$$\{u\}_{\Gamma_2} = \{u\}_{\Gamma_1} = \{u\}_{\Gamma} \quad \{r\}_{\Gamma_2} = -\{r\}_{\Gamma_1} \quad (1.54)$$

where it is assumed that the interface DOFs have been sorted giving the same index to corresponding DOFs of corresponding nodes. Substituting equations (1.54) back into equation (1.53) and eliminating the residuals one can get:

$$\begin{bmatrix} [\mathbf{K}]_{1,1} & [\mathbf{K}]_{1,\Gamma_1} & [0] & [0] \\ [\mathbf{K}]_{\Gamma_1,1} & [\mathbf{K}]_{\Gamma_1,\Gamma_1} + [\mathbf{K}]_{\Gamma_2,\Gamma_2} & [\mathbf{K}]_{\Gamma_2,2} & [\mathbf{K}]_{2,2} \\ [0] & [\mathbf{K}]_{2,\Gamma_2} & [\mathbf{K}]_{2,2} & \end{bmatrix} \begin{Bmatrix} \{u\}_1 \\ \{u\}_{\Gamma} \\ \{u\}_2 \end{Bmatrix} = \begin{Bmatrix} \{f\}_1 \\ \{f\}_{\Gamma_1} + \{f\}_{\Gamma_2} \\ \{f\}_2 \end{Bmatrix} \quad (1.55)$$

Obviously, in the case of a conforming interface equation (1.55) could also be obtained using the conforming discretization of equation (1.13). Indeed equations (1.32) and (1.55) are equivalent.

### 1.3.4 Domain decomposition with non-consistent discretization

Cafiero et al. [Cafiero et al., 2016] shows different cases in which the non-conforming discretization can increase the challenge associated with a "transfer" of displacement field and stress tensor between meshes. In the following section the different scenarios are reviewed.

#### 1.3.4.1 Non conformity scenarios

The first case addressed here is the simplest one (cf. figure 1.8). The interface boundary and area are the same,  $\Gamma_1^h \equiv \Gamma_2^h$ , the normal vectors are opposite in each corresponding point of the interfaces. It may be the case, in the context of domain decomposition for a multi-grid approach. The nodes are not in the same position but the shape functions of both domains are still the same. As a consequence, the local balance of the surface traction using a mortar-type approach will also imply the satisfaction of a patch test [Park et al., 2002].

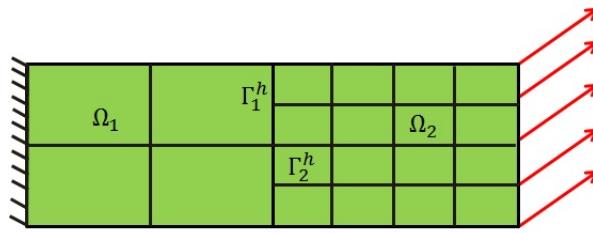


Figure 1.8: Inconsistent mesh partition that does not introduce geometric inconsistency:  $\Gamma_1^h \equiv \Gamma_2^h$

The second case is the first case of geometric inconsistency due to the non-coincidence of the boundary of the interface (figure 1.9).

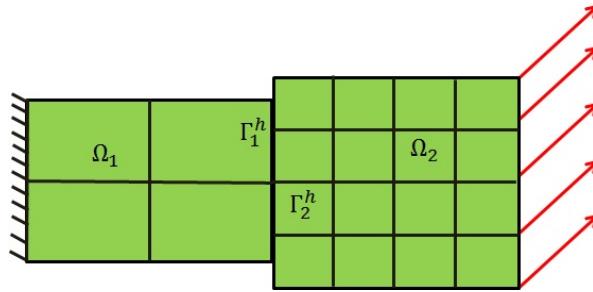


Figure 1.9: Inconsistent mesh partition that introduces geometric inconsistency:  $\Gamma_1^h \neq \Gamma_2^h$ , first case different mesh boundaries

Still the surface normal is the opposite that means that the patch test may be satisfied by the use of a mortar approach. In figure 1.10 a second case that also produces a geometric inconsistency is presented. When a curved interface is discretized with non-matching meshes,

the normal vectors are distinct on each element surface and may not be opposite for corresponding elements<sup>6</sup>. The area is different on both mesh surfaces. Moreover there can be a gap or intersections of interface mesh surfaces.

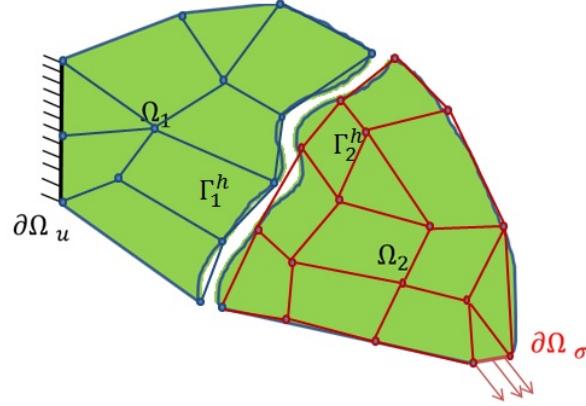


Figure 1.10: Inconsistent mesh partition that generates geometric inconsistency:  $\Gamma_1^h \not\equiv \Gamma_2^h$ , second case curbed interface

In this case the satisfaction of the patch test is only approximated with the use of a mortar approach [Park et al., 2002]. Another important scenario is the one in figure 1.11 that shows two domains with the same node position but with different shape functions. Also in this case

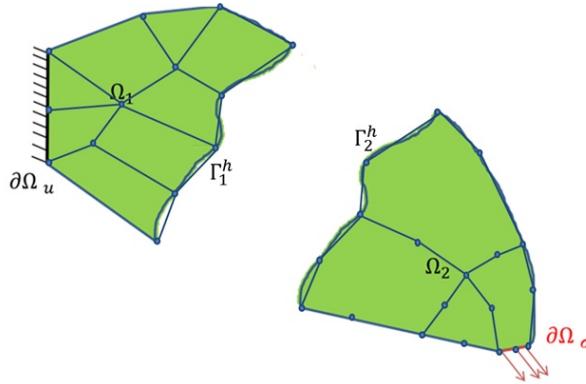


Figure 1.11: Inconsistent mesh partition that introduces geometric inconsistency:  $\Gamma_1^h \not\equiv \Gamma_2^h$ , third case different shape functions

the normal and the geometry of the interpolated interfaces are not coincident, and special methods are needed to deal with this case [Wang et al., 2017]. The most general scenario could be the combination of one or more of these scenarios, as it will be discussed later.

<sup>6</sup>In this case even the definition of corresponding elements is not straightforward. A possible definition is that two elements are corresponding if the area of the intersection between one element surface and the projection of the other element surface on the first one is non-zero and if the element center distance is less than a given tolerance

### 1.3.5 Elimination methods - Master/Slave approaches

This is a family of approaches that is described in Barlow et al. [Barlow, 1982] and in the review paper of De Boer et al [De Boer et al., 2007] that is briefly summarized and enriched in this section. The main idea consists in the reformulation of equations (1.49) and (1.50) as linear relationships between the interfaces of the sub-domains. Then, one of the surfaces is chosen as slave ( $\Gamma_2$ ), its DOFs will be explicitly given as function of the DOFs of the master surface ( $\Gamma_1$ ).

$$\{u\}_{\Gamma_2} = [\Pi]_{21} \{u\}_{\Gamma_1} \quad (1.56)$$

From a physical point of view the total work as the resultant force of the internal forces on the interface should be zero (energy conservation and static balance).

$$\{r\}_{\Gamma_1}^T \{u\}_{\Gamma_1} + \{r\}_{\Gamma_2}^T \{u\}_{\Gamma_2} = 0 \quad (1.57)$$

$$[1]_{\Gamma_1}^T \{r\}_{\Gamma_1} + [1]_{\Gamma_2}^T \{r\}_{\Gamma_2} = \{0\} \quad (1.58)$$

In which  $[1]_{\Gamma_1}$  and  $[1]_{\Gamma_2}$  are matrices with the number of lines equal to the number of DOFs of interface  $\Gamma_1$  and  $\Gamma_2$  and 3 columns (one per each component). For 3D solid elements they are Boolean matrices that have 1 on the component whose line corresponds to the direction given by the index of their column and 0 otherwise. Equations (1.57) and (1.58) combined with equation (1.56) give:

$$\{r\}_{\Gamma_1} = -[\Pi]_{21}^T \{r\}_{\Gamma_2} \quad (1.59)$$

$$[\Pi]_{21} [1]_{\Gamma_1} = [1]_{\Gamma_2} \quad (1.60)$$

Equation (1.60) is a necessary condition that must be satisfied by the projection operator  $[\Pi]_{21}$  defined in equation (1.56). On the other hand, equation (1.59) is used with equations (1.53) and (1.54) to finally get the reduced system of equations:

$$\begin{bmatrix} [\mathbf{K}]_{1,1} & [\mathbf{K}]_{1,\Gamma_1} & [0] \\ [\mathbf{K}]_{\Gamma_1,1} & [\mathbf{K}]_{\Gamma_1,\Gamma_1} + [\Pi]_{21}^T [\mathbf{K}]_{\Gamma_2,\Gamma_2} [\Pi]_{21} & [\Pi]_{21}^T [\mathbf{K}]_{\Gamma_2,2} \\ [0] & [\mathbf{K}]_{2,\Gamma_2} [\Pi]_{21} & [\mathbf{K}]_{2,2} \end{bmatrix} \begin{Bmatrix} \{u\}_1 \\ \{u\}_{\Gamma_1} \\ \{u\}_2 \end{Bmatrix} = \begin{Bmatrix} \{f\}_1 \\ \{f\}_{\Gamma_1} + [\Pi]_{21}^T \{f\}_{\Gamma_2} \\ \{f\}_2 \end{Bmatrix} \quad (1.61)$$

This is a linear system of equations with a positive semi-definite symmetric matrix like the stiffness matrix of equation (1.32). The projection operator  $[\Pi]_{21}$  can be defined using two different approaches that in the contact community are known as node to segment and segment to segment approaches. In the former the relation between the DOFs is given directly by a geometric interpolation. De Boer et al. [De Boer et al., 2007] show two examples of these methods: the Nearest neighbor interpolation (NN) [Thévenaz et al., 2000] and the Radial Basis Functions (RBF) [Beckert and Wendland, 2001], [Smith et al., 2000]. Another option consists in using of the trace of element shape function on the interface to interpolate the displacement field. In this work these methods are referred to as Element Shape Function interpolation (ESF). The main drawback of these methods is their strong dependence on the slave and master surface choice. In fact for domains meshed with different mesh sizes, choosing the most refined mesh as master surface will not ensure the continuity of the displacement field

for all the master nodes that do not belong to elements that contain a slave node projection. Moreover these methods don't satisfy simple patch tests even if the interface is not curved, and finally as it was shown by Bernardi et al. [Bernardi, 1989] they are sub-optimal in term of mesh convergence. The segment to segment approaches even if much more complex to set up can show better accuracy. Finally another elimination method based on a weak form of the continuity equation is presented: the Weighted Residual Methods (WRM).

### 1.3.5.1 Radial Basis Function interpolation (RBF)

For more general applications, the RBF represent a practical and easy to implement solution. In this approach the displacement field at the interface is supposed to be a linear combination of radial basis functions defined per each node of the master surface:

$$\{u\}_{\Gamma_1}(\{x\}) = \sum_{s=1}^{N_{\Gamma_1}} \alpha_s \phi_s(\|\{x\} - \{x\}_s^{\Gamma_1}\|, r_s) + p(\{x\}) \quad (1.62)$$

Here  $\{u\}_{\Gamma_1}$  is the displacement field on surface  $\Gamma_1$ .  $\alpha_s$  is the weight relative to the s-th radial basis function  $\phi_s$ ,  $\|\{x\} - \{x\}_s^{\Gamma_1}\|$  is the Euclidean distance of point  $\{x\}$  from the s-th node of the master surface  $\Gamma_1$ ,  $r_s \in \mathbb{R}$  is the shape factor of the s-th radial basis function  $\phi_s$ ,  $N_{\Gamma_1}$  is the number of nodes of  $\Gamma_1$  and  $p(\{x\})$  is a polynomial function of  $\{x\}$ . In this work  $p(\{x\})$  is not considered, for a more general discussion of RBF the reader is referred to [De Boer et al., 2007], [Beckert and Wendland, 2001] and [Smith et al., 2000]. The weights  $\alpha_s$  of equation (1.62) can be found using the interpolation conditions:

$$\{u\}_{\Gamma_1}(\{x\}_i^{\Gamma_1}) = u_{\Gamma_1 i} \quad \forall i = 1, \dots, N_{\Gamma_1} \quad (1.63)$$

As a consequence the weights factors will be

$$[A] = [\Phi]^{-1} \{u\}_{\Gamma_1} \quad (1.64)$$

Where  $[\Phi]$  was defined as:

$$\Phi_{i,j} = \phi_i(\|\{x\}_j^{\Gamma_1} - \{x\}_i^{\Gamma_1}\|, r_s) \quad (1.65)$$

Finally to impose the continuity of displacements at the interface (equation (1.54)) one can use the interpolation given by the RBF that takes the form:

$$\{u\}_{\Gamma_2} = [\Psi]^T \cdot [A] = [\Psi]^T \cdot [\Phi]^{-1} \{u\}_{\Gamma_1} = [\Pi]_{21} \{u\}_{\Gamma_1} \quad (1.66)$$

Here the matrix  $\Psi$  is defined as:

$$\Psi_{i,j} = \phi_i(\|\{x\}_j^{\Gamma_2} - \{x\}_i^{\Gamma_1}\|, r_s) \quad (1.67)$$

Different radial basis functions ( $\phi(\cdot)$ ) can be found in the literature [Wendland, 1995], [Buhmann, 2003], [Duchon, 1977]. An example considered in this work is the compactly supported

function B&W [Wendland, 1995]:

$$\phi(\|\{x\}\|, r_s) = \left(1 - \frac{\|\{x\}\|}{r_s}\right)_+^4 \left(1 + 4 \frac{\|\{x\}\|}{r_s}\right) \quad (1.68)$$

A benchmark of the most popular RBF for nodal interpolation can be found in Deparis et al. [Deparis et al., 2014]. The latter deals with the choice of the shape factor  $r_s$  in a local manner in order to find a good compromise between precision and evaluation cost ([ $\Phi$ ] band width). In the same work the RBFs are rescaled in order to avoid spurious oscillations due to interpolation so that they satisfy the condition given by equation (1.60).

### 1.3.5.2 Element Shape Function interpolation (ESF)

A natural, but more complex, approach consists in the projection of slave surface nodes on master surface elements, so that :

$$\Pi_{21i,j} = [N]_j^{\Gamma_1} (\{x\}_i^{\Gamma_2}) \quad (1.69)$$

Where  $[N]_j^{\Gamma_1}$  is the j-th column of the  $[N]^{(h)}(\{x\})$  matrix of equation (1.15) for  $\Gamma_1$  interface elements. This projection is quite simple for planar interfaces but for general curved interfaces the projection may be more complicated. An elegant solution is presented in Puso et al. [Puso, 2004] who considered the normal vector in the center of master elements to rapidly evaluate the projection. For nearly planar surfaces this can be a reasonable approximation, on the other hand, for double curved surfaces this approximation may not be justified. For each node ( $i$ ) of  $\Gamma_1$  and for each element ( $j$ ) of  $\Gamma_2$ , the local coordinates  $(\xi_{ij}, \eta_{ij})$  that describe the position of the node ( $i$ ) on the surface  $\Gamma_j$  described by the element ( $j$ ) need to be determined. In the case of geometrical inconsistency (figure 1.10) the node ( $i$ ) may not lie on the surface  $\Gamma_j$ , in that case the local coordinates of the point of  $(i^*) \in \Gamma_j$  ( as in figure 1.12) that is the closest to ( $i$ ) is to be determined i.e.

$$\min_{(\xi_j, \eta_j)} d_{ij}^2(\xi_j, \eta_j) = \min_{(\xi_j, \eta_j)} \left( \|\{x\}(\xi_j, \eta_j) - \{x\}_{(i)}\|^2 \right) \quad (1.70)$$

To find all the corresponding local coordinates of one mesh projected on the other this optimization problem has to be solved for all  $N_{\Gamma_1}^{(e)} \times N_{\Gamma_2}$  pairs of nodes ( $i$ ) element ( $j$ ). To improve the numerical efficiency of this procedure the squared distance function may be summed up to build a function that has to be minimized once for each node ( $i$ ) to get its local coordinate in each corresponding element  $\{\Xi\}, \{\Theta\}$ :

$$D_i^2(\{\Xi\}, \{\Theta\}) = \sum_{j=1}^{N_{\Gamma_1}^{(e)}} d_{ij}^2(\xi_j, \eta_j) \quad (1.71)$$

Moreover the gradient of each squared distance can be computed analytically to speed up the optimization convergence using a gradient based optimization algorithm for unconstrained

problems for example a Quasi-Newton algorithm [Dennis and Moré, 1977, Davidon, 1959]-[Fletcher and Powell, 1963, Dennis Jr and Schnabel, 1996].

$$\nabla d_{i,j}^2(\xi_j, \eta_j) = 2\nabla \{x\}(\xi_j, \eta_j) \cdot (\{x\}(\xi_j, \eta_j) - \{x\}_{(i)}) \quad (1.72)$$

When this procedure is used, for convex interface meshes there may be nodes that are far

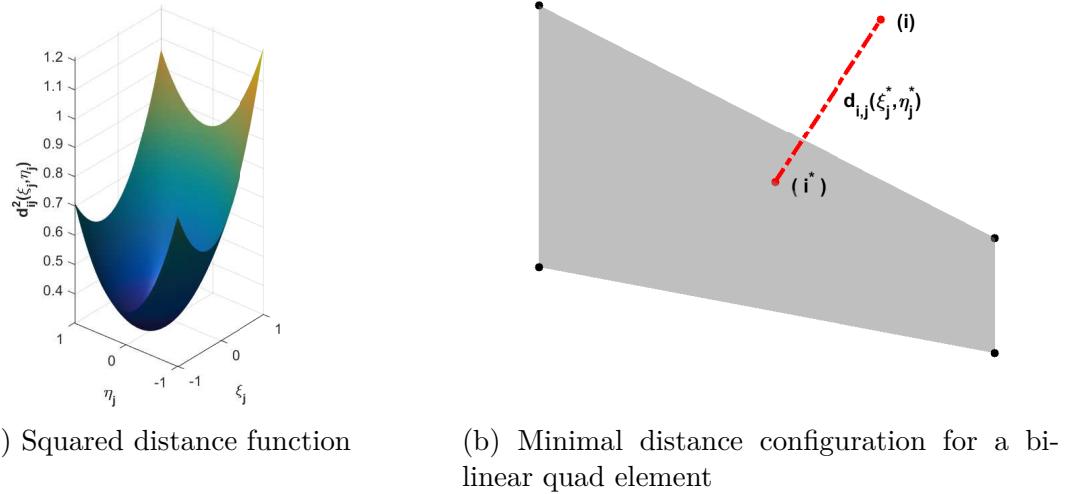


Figure 1.12: Node to element projection

from the element ( $j$ ) but that have their corresponding point ( $i^*$ ) inside the element ( $j$ ). The shape function of equation (1.69) should take in account the minimal distance  $d_{ij}(\xi_j^*, \eta_j^*)$ : all the DOFs of the node ( $i$ ) at a distance superior to a given tolerance  $t$  to its projection ( $i^*$ ) on  $\Gamma_j$ , should not be considered dependent on the ( $j$ ) elements DOFs i.e.:

$$\Pi_{21i,j} = \begin{cases} [N]_j^{\Gamma_1}(\xi^*, \eta^*) & d_{ij}(\xi_j^*, \eta_j^*) \leq t \\ 0 & \text{otherwise.} \end{cases} \quad (1.73)$$

### 1.3.5.3 Weighted Residual Methods (WRM)

The methods described until now are based on a strong formulation of the continuity on the interface of the displacement field (equation (1.56)). On the other hand, segment to segment methods try to impose the continuity of displacements at the interface by the use of a weak form. One such method is the Weighted residual methods described in De Boer et al. [De Boer et al., 2007], Cebral et al [Cebral and Löhner, 1997] and Löhner et al. [Löhner et al., 1998] for the fluid-structure interaction problem.<sup>7</sup> In this approach the jump of the

<sup>7</sup>These approaches do not make the use of Lagrange multipliers like in the Mortar methods ([Bernardi, 1989], [Bernardi et al., 1993]); anyway when the shape functions of Lagrange multipliers are the same of the displacement field of the slave surface, the consequent equations are the same as it is also shown in Jeon et al. [Jeong et al., 2017]

displacement field across the interface has to be orthogonal in the  $\mathbb{L}_2$  sense to the trace on  $\Gamma_2$  of the kinematic admissible virtual displacement  $\{v\} \in \mathbb{V}$ :

$$\int_{\Gamma} \{v\}(\{x\}) \cdot (\{u\}_{\Gamma_2}(\{x\}) - \{u\}_{\Gamma_1}(\{x\})) d\Gamma = 0 \quad \forall \{v\} \in \mathbb{V} \quad (1.74)$$

Discretizing  $\{v\}(\{x\})$  using the shape functions of  $\Gamma_2$  and using equation (1.14) for each sub-domains and writing one equation for each shape function of the discretized space  $\mathbb{V}_{\Gamma_2}^h$  equation (1.74) becomes :

$$[M]_2 \{u\}_{\Gamma_2} - [M]_{21} \{u\}_{\Gamma_1} = \{0\} \quad (1.75)$$

where the mass matrices are defined as:

$$M_{2i,j} = \int_{\Gamma} [N]_i^{\Gamma_2}(\{x\}) \cdot [N]_j^{\Gamma_2}(\{x\}) d\Gamma \quad (1.76)$$

$$M_{21i,j} = \int_{\Gamma} [N]_i^{\Gamma_2}(\{x\}) \cdot [N]_j^{\Gamma_1}(\{x\}) d\Gamma \quad (1.77)$$

The slave DOFs can be written as a function of the master DOFs as:

$$\{u\}_{\Gamma_2} = [M]_2^{-1} [M]_{21} \{u\}_{\Gamma_1} = [\Pi]_{21} \{u\}_{\Gamma_1} \quad (1.78)$$

It is shown in De Boer et al. [De Boer et al., 2007] that this method respects the condition of equation (1.60), therefore the force resultant is conserved at the interface. In order to evaluate the integral from equation (1.76) and (1.77) the support of the integral has to be chosen.

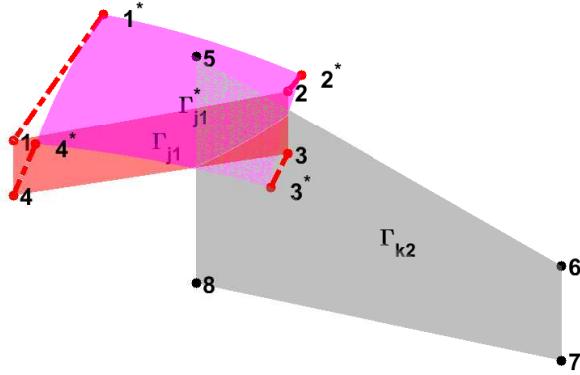
In fact in the most general case the surfaces described by the meshes will not be coincident and a projection will be needed (cf. figure (1.13))<sup>8</sup>

One can observe that the integrand's support (cf. figure 1.14c) is  $\Gamma_{j1^*k2} \equiv \Gamma_{j1^*} \cap \Gamma_{k2}$ . The total support  $\Gamma$  is therefore the union of all the intersections of each  $\Gamma_{k2}$  with all corresponding projections  $\Gamma_{j1^*}$  of  $\Gamma_{j1}$  on  $\Gamma_{k2}$ :

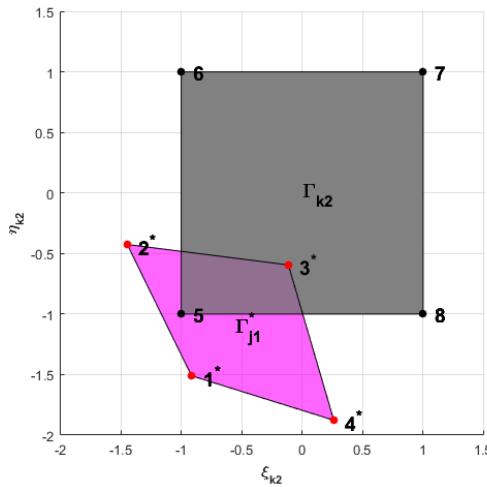
$$\Gamma \equiv \bigcup_{k2=1}^{N_2^{(e)}} \bigcup_{j1=1}^{N_1^{(e)}} \Gamma_{j1^*k2} \quad (1.79)$$

For the numeric evaluation of  $[M]_2$  and  $[M]_{21}$  the proposed implementation is inspired by the procedure in Puso et al. [Puso, 2004]. However compared to Puso's original, some modifications were implemented in terms of node-to-element projection, and in terms of intersection polygon definition in order to render the procedure more robust. Instead of using the clipping algorithm of Foley [D and Van, 1996] that may fail in some cases ( see [Gander and Japhet, 2013] for further details), the procedure described by Gander et al. in [Gander and Japhet, 2013] was implemented. To obtain the intersection polygon  $\Gamma_{j1^*k2}$ , firstly the nodes ( $s1$ ) of  $\Gamma_{j1}$  that have a projection inside the element ( $k2$ ) are found. Secondly the nodes ( $s2$ ) of  $\Gamma_{k2}$  that are inside the projected element surface  $\Gamma_{j1^*}$  are found. Thirdly the intersection points

<sup>8</sup>Note that  $\Gamma_{j1}^*$  sides are generally curved, in this work they are considered to be straight; this approximation will affect the accuracy of the evaluation of  $\mathbf{M}_{21}$  in equation (1.77). Nevertheless the error induced by this approximation can be considered negligible.



(a) Elements belonging to inconsistent meshes of the same interface surface.



(b) Projected master element on slave element surface in slave local coordinates.

Figure 1.13: Element to element projection examples. The master element surface  $\Gamma_{j1}$  has been projected on the slave element surface  $\Gamma_{k2}$  using the same procedure shown in figure (1.12) for each node of the element  $j1$  ( $1, 2, 3, 4$ ). In this way it is possible to obtain the projected slave surface  $\Gamma_{j1}^*$  ( $1^*, 2^*, 3^*, 4^*$ ). The union of the projection of all the elements of  $\Gamma_1$  on  $\Gamma_2$  is then indicated as  $\Gamma_1^*$ . After this projection, a change in local variables  $(\xi_{k2}, \eta_{k2})$  of the element  $k2$  can be employed for the evaluation of Mass matrices (equations (1.76) and (1.77))

(s3) of all the sides of  $\Gamma_{k2}$  with the one of  $\Gamma_{j1}^*$  is carried out. The union of (s1), (s2) and (s3) form the vertices of the intersection polygon see figure 1.15a.

A triangulation of this polygon is obtained connecting all the sides with the polygon center.

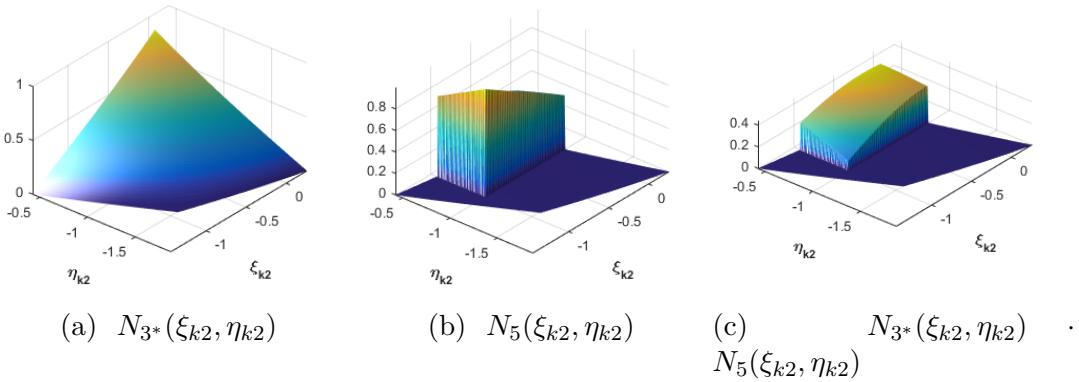


Figure 1.14: Shape function of node  $3^*$  of  $j1$ , of node 5 of  $k2$  for the example of figure 1.13 and their product that has to be integrated in equation (1.77), are represented.

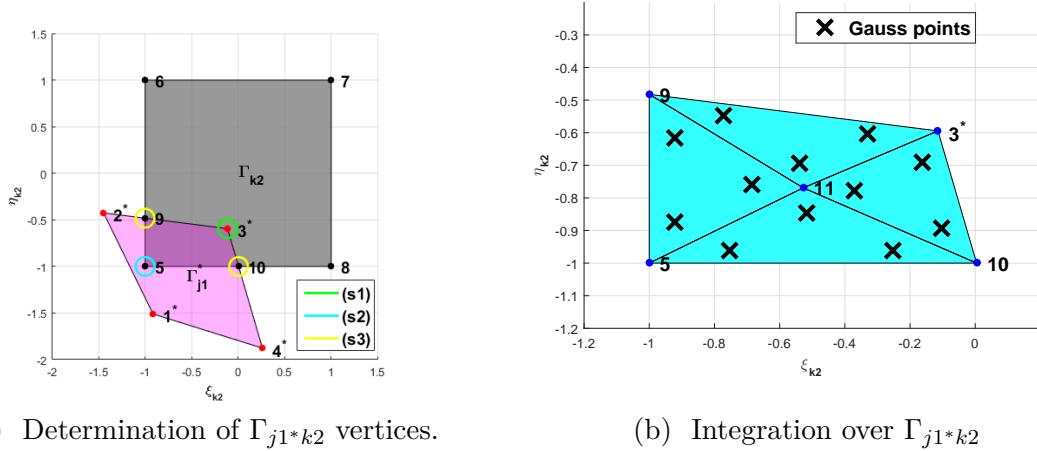


Figure 1.15: Procedure for the determination of  $\Gamma_{j1^*k2}$  vertices and consequent numerical integration

Finally, 3 Gauss points per triangle have to be used for the numerical integration of equation (1.76) and (1.77)) (see the example of figure 1.15b). The Gauss point local coordinates ( $\{\Xi\}_{k2}^{(GP)}, \{\Theta\}_{k2}^{(GP)}$ ) are easy to find as a function of the triangulation coordinates. On the other hand, to know the value of  $[N]_j^{\Gamma_1}$  on the corresponding points of  $\Gamma_{j1^*}$  for equation (1.77)), the local coordinates ( $\{\Xi\}_{j1}^{(GP)}, \{\Theta\}_{j1}^{(GP)}$ ) have to be determined. Puso suggested even for this projection to use the normal of element ( $k2$ ) on the center of the element. To be completely rigorous one should evaluate the normal to the surface  $\Gamma_{k2}$  in each Gauss point and then make an intersection of the line on which the normal lies and the surface of  $\Gamma_{j1}$  to finally have the local coordinates ( $\{\Xi\}_{j1}^{(GP)}, \{\Theta\}_{j1}^{(GP)}$ ). In other words one should find the

roots of the nonlinear system of equations:

$$2\nabla \{x\}_{\Gamma_{k2}} \left( \{\Xi\}_{k2}^{(GP)}, \{\Theta\}_{k2}^{(GP)} \right) \cdot \\ \cdot \left( \{x\}_{\Gamma_{k2}} \left( \{\Xi\}_{k2}^{(GP)}, \{\Theta\}_{k2}^{(GP)} \right) - \{x\}_{\Gamma_{j1}} \left( \{\Xi\}_{j1}^{(GP)}, \{\Theta\}_{j1}^{(GP)} \right) \right) = \{0\} \quad (1.80)$$

Taking in consideration the fact that the sides of  $\Gamma_{j1*}$  are considered as straight segment between the projection on  $\Gamma_{k2}$  of the nodes of ( $j1$ ), even if this is not always the case, the use of equation (1.80) is not recommended. Some Gauss points when re-projected on  $\Gamma_{j1}$  could lie outside the element ( $j1$ ). To avoid this situation a geometric isoparametric interpolation between the local coordinate  $(\xi_{j1}, \eta_{j1})$  and the projected one  $(\xi_{k2}, \eta_{k2})$  is considered here:

$$\xi_{k2}(\xi_{j1}, \eta_{j1}) = \xi_{k2}^{1*}N_1(\xi_{j1}, \eta_{j1}) + \xi_{k2}^{2*}N_2(\xi_{j1}, \eta_{j1}) + \xi_{k2}^{3*}N_3(\xi_{j1}, \eta_{j1}) + \xi_{k2}^{4*}N_4(\xi_{j1}, \eta_{j1}) \quad (1.81)$$

$$\eta_{k2}(\xi_{j1}, \eta_{j1}) = \eta_{k2}^{1*}N_1(\xi_{j1}, \eta_{j1}) + \eta_{k2}^{2*}N_2(\xi_{j1}, \eta_{j1}) + \eta_{k2}^{3*}N_3(\xi_{j1}, \eta_{j1}) + \eta_{k2}^{4*}N_4(\xi_{j1}, \eta_{j1}) \quad (1.82)$$

Where  $(\xi_{k2}^{n*}, \eta_{k2}^{n*})$  are the local coordinates of the projection of the n-th node of ( $j1$ ) on  $\Gamma_{k2}$  and  $N_n$  are the classic bilinear shape functions:

$$\begin{aligned} N_1(\xi_{j1}, \eta_{j1}) &= \frac{1}{4}(1 - \xi_{j1})(1 - \eta_{j1}) \\ N_2(\xi_{j1}, \eta_{j1}) &= \frac{1}{4}(1 - \xi_{j1})(1 + \eta_{j1}) \\ N_3(\xi_{j1}, \eta_{j1}) &= \frac{1}{4}(1 + \xi_{j1})(1 + \eta_{j1}) \\ N_4(\xi_{j1}, \eta_{j1}) &= \frac{1}{4}(1 + \xi_{j1})(1 - \eta_{j1}) \end{aligned} \quad (1.83)$$

Finally, equation (1.81)-(1.82) can be used to find  $(\{\Xi\}_{j1}^{(GP)}, \{\Theta\}_{j1}^{(GP)})$  solving another non-linear system of equations:

$$\{\Xi\}_{k2}^{(GP)} - \xi_{k2} \left( \{\Xi\}_{j1}^{(GP)}, \{\Theta\}_{j1}^{(GP)} \right) = \{0\} \quad (1.84)$$

$$\{\Theta\}_{k2}^{(GP)} - \eta_{k2} \left( \{\Xi\}_{j1}^{(GP)}, \{\Theta\}_{j1}^{(GP)} \right) = \{0\} \quad (1.85)$$

This system can be solved numerically (e.g. with a Newton-Raphson algorithm) giving the analytical expression of the Jacobian matrix. In figure 1.16 the Gauss points are projected from  $\Gamma_{j1*k2}$  back on  $\Gamma_{j1}$  using the iso-parametric projection of equations (1.84)-(1.85).

### 1.3.6 Variational based approaches

In variational approaches, instead of using the principle of virtual work to get the final system of equations, the displacement field solution of the static problem will be sought to minimize the total energy:

$$E(\{u\}) = \{u\}^T [K] \{u\} - \{u\}^T \{f\} \quad (1.86)$$

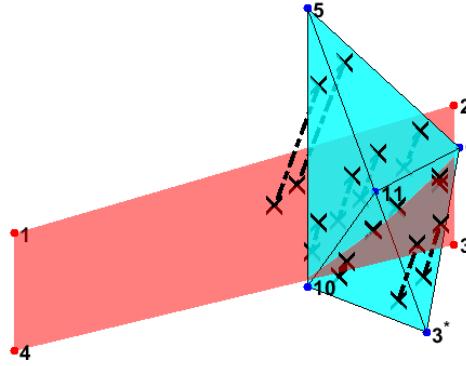


Figure 1.16: Gauss point projection from  $\Gamma_{j1*k2}$  back on  $\Gamma_{j1}$  using the iso-parametric projection of equations (1.84)-(1.85).

Considering the aforementioned partition, the total energy becomes:

$$E(\{u\}_1, \{u\}_{\Gamma_1}, \{u\}_{\Gamma_2}, \{u\}_2) = \left\{ \begin{array}{c} \{u\}_1^T \quad \{u\}_{\Gamma_1}^T \quad \{u\}_{\Gamma_2}^T \quad \{u\}_2^T \end{array} \right\} \left( \begin{bmatrix} [K]_{1,1} & [K]_{1,\Gamma_1} & [0] & [0] \\ [K]_{\Gamma_1,1} & [K]_{\Gamma_1,\Gamma_1} & [0] & [0] \\ [0] & [0] & [K]_{\Gamma_2,\Gamma_2} & [K]_{\Gamma_2,2} \\ [0] & [0] & [K]_{2,\Gamma_2} & [K]_{2,2} \end{bmatrix} \begin{Bmatrix} \{u\}_1 \\ \{u\}_{\Gamma_1} \\ \{u\}_{\Gamma_2} \\ \{u\}_2 \end{Bmatrix} - \begin{Bmatrix} \{f\}_1 \\ \{f\}_{\Gamma_1} \\ \{f\}_{\Gamma_2} \\ \{f\}_2 \end{Bmatrix} \right) \quad (1.87)$$

Note that in equation (1.87) the work at the interface of the residual is eliminated for energy conservation. The system of equations that comes from this formulation has a singular matrix, so that the solution of the static problem cannot be simply obtained by minimizing the total energy.

### 1.3.6.1 Mortar Element Method

The basic idea of the Mortar approach is to add a dislocation potential to the total energy, in order to impose the continuity of the displacement field at the interface:

$$E_d(\{u\}, \{\lambda\}) = \int_{\Gamma} \{\lambda\} (\{u\}|_{\Gamma_1} - \{u\}|_{\Gamma_2}) d\Gamma \quad (1.88)$$

This potential is summed with the total energy to get the Lagrangian functional  $E_l(\{u\}, \{\lambda\}) = E_d(\{u\}, \{\lambda\}) + E(\{u\})$  whose stationary points are the solution of the constrained optimization:

$$\begin{cases} \min_{\{u\}} (E(\{u\})) \\ \{u\}|_{\Gamma_1} - \{u\}|_{\Gamma_2} = 0 \end{cases} \quad (1.89)$$

Similarly to Master/Slave approaches, the mortar approach needs the choice of one surface to be the mortar surface and the other to be the non-mortar. The interpolation functions of the

Lagrangian multipliers  $\{\lambda\}$  are (in the case of two domains) chosen to be the same as those of the slave surface:<sup>9</sup>

$$\{\lambda\}(\{x\}|_{\Gamma_2}) = [N](\{x\}|_{\Gamma_2})\{\lambda\}_{\Gamma_2} \quad (1.90)$$

The dislocation potential can then be written as:

$$E_d(\{u\}_{\Gamma_1}, \{u\}_{\Gamma_2}, \{\lambda\}_{\Gamma_2}) = \{\lambda\}_{\Gamma_2}^T ([M]_{21}\{u\}_{\Gamma_1} - [M]_2\{u\}_{\Gamma_2}) \quad (1.91)$$

Then the Lagrangian functional is stationary for:

$$\begin{bmatrix} [\mathbf{K}]_{1,1} & [\mathbf{K}]_{1,\Gamma_1} & [0] & [0] & [0] \\ [\mathbf{K}]_{\Gamma_1,1} & [\mathbf{K}]_{\Gamma_1,\Gamma_1} & [M]_{21}^T & [0] & [0] \\ [0] & [M]_{21} & [0] & -[M]_2 & [0] \\ [0] & [0] & -[M]_2^T & [\mathbf{K}]_{\Gamma_2,\Gamma_2} & [\mathbf{K}]_{\Gamma_2,2} \\ [0] & [0] & [0] & [\mathbf{K}]_{2,\Gamma_2} & [\mathbf{K}]_{2,2} \end{bmatrix} \begin{Bmatrix} \{u\}_1 \\ \{u\}_{\Gamma_1} \\ \{\lambda\}_{\Gamma_2} \\ \{u\}_{\Gamma_2} \\ \{u\}_2 \end{Bmatrix} = \begin{Bmatrix} \{f\}_1 \\ \{f\}_{\Gamma_1} \\ \{0\} \\ \{f\}_{\Gamma_2} \\ \{f\}_2 \end{Bmatrix} \quad (1.92)$$

Even if the equations obtained from this approach seem to be different from the one of the WRM, eliminating the Lagrange eigenvalues from equation (1.92) it is possible to show that they have the same solution (see Jeong et al. [[Jeong et al., 2017](#)] for the proof). It can be noted that this method has the same difficulties of implementation and of evaluation cost encountered with the WRM method: The integral of the shape function over the intersection of the interface has to be evaluated in order to evaluate the mass matrices  $[M]_{21}, [M]_2$ . The Mortar approach is shown to be as accurate as the WRM approach but with a noticeably longer effort due to the increased number of variables. For this reason in section 1.3.10 only WRM has been implemented and compared with other methods.

### 1.3.7 The Internodes Approach

Here the internodes approach, introduced by Deparis et al in [[Deparis et al., 2016](#)] and further analyzed by Gervasio et al. in [[Gervasio and Quarteroni, 2016](#)], is described. Its matrix formulation is derived directly from equation (1.51). Like in the elimination approaches the continuity of the displacement field is guaranteed by equation (1.56). On the other hand the balance of energy and force sum are not imposed. The forces  $\{t\}_{12}$  and  $\{t\}_{21}$  are supposed to be interpolated with the same shape functions of displacement on each subdomain interface:

$$\{t\}_{12}(\{x\}|_{\Gamma_2}) = [N]^{(h)}(\{x\}|_{\Gamma_2})\{p\}_2 \quad (1.93)$$

$$\{t\}_{21}(\{x\}|_{\Gamma_1}) = [N]^{(h)}(\{x\}|_{\Gamma_1})\{p\}_1 \quad (1.94)$$

where  $\{p\}_1$  and  $\{p\}_2$  are the vector of internal forces per unit of surface interpolating  $\{t\}_{12}(\{x\}|_{\Gamma_2})$  and  $\{t\}_{21}(\{x\}|_{\Gamma_1})$ . Conforming to these definitions the residual at the interface may be eval-

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<sup>9</sup>In the work of Puso et al [[Puso, 2004](#)] the dual space is also employed

uated as:

$$\{r\}_{\Gamma_1} = \int_{\Gamma_1} [N]^{(h)} (\{x\}|_{\Gamma_1}) \cdot [N]^{(h)} (\{x\}|_{\Gamma_1}) \{p\}_1 d\Gamma_1 = [M]_1 \{p\}_1 \quad (1.95)$$

$$\{r\}_{\Gamma_2} = \int_{\Gamma_2} [N]^{(h)} (\{x\}|_{\Gamma_2}) \cdot [N]^{(h)} (\{x\}|_{\Gamma_2}) \{p\}_2 d\Gamma_2 = [M]_2 \{p\}_2 \quad (1.96)$$

The balance of the internal forces per unit area is imposed in a similar fashion to the continuity of the displacements as in equation (1.56), but this time a second interpolation operator is used:

$$\{p\}_1 + [\Pi]_{12} \{p\}_2 = \{0\} \quad (1.97)$$

Substituting equation (1.95) and (1.96) in (1.97) one gets:

$$\{r\}_{\Gamma_1} + [M]_1 [\Pi]_{12} [M]_2^{-1} \{r\}_{\Gamma_2} = \{r\}_{\Gamma_1} + [Q]_{12} \{r\}_{\Gamma_2} = \{0\} \quad (1.98)$$

Finally, using equations (1.98) and (1.56) in (1.51) one can get:

$$\begin{bmatrix} [\mathbf{K}]_{1,1} & [\mathbf{K}]_{1,\Gamma_1} & [0] \\ [\mathbf{K}]_{\Gamma_1,1} & [\mathbf{K}]_{\Gamma_1,\Gamma_1} + [Q]_{12} [\mathbf{K}]_{\Gamma_2,\Gamma_2} [\Pi]_{21} & [Q]_{12} [\mathbf{K}]_{\Gamma_2,2} \\ [0] & [\mathbf{K}]_{2,\Gamma_2} [\Pi]_{21} & [\mathbf{K}]_{2,2} \end{bmatrix} \begin{Bmatrix} \{u\}_1 \\ \{u\}_{\Gamma_1} \\ \{u\}_2 \end{Bmatrix} = \begin{Bmatrix} \{f\}_1 \\ \{f\}_{\Gamma_1} + [Q]_{12} \{f\}_{\Gamma_2} \\ \{f\}_2 \end{Bmatrix} \quad (1.99)$$

It can be observed that the system of equations of the internodes formulation is not symmetric and that equations (1.59) and (1.60) are not imposed, so that internodes does not conserve *a priori* neither work nor resultants of forces and moments. On the other hand the formulation and the implementation are straightforward if compared with the mortar and the WRM approaches: only the mass matrices  $[M]_2$ ,  $[M]_1$  and the interpolation operators  $[\Pi]_{12}$  and  $[\Pi]_{21}$  are needed. Good convergence properties have been showed in the work of Gervasio et al. [[Gervasio and Quarteroni, 2016](#)].

### 1.3.8 Weighted Average Continuity Approach (WACA)

The collocation approaches even if simple are not accurate enough when the most refined mesh is chosen as master surface. On the other hand the segment to segment approaches are very accurate but are complex and need much more computational effort. The Internodes approach seeks to achieve such a trade-off between complexity and accuracy but, as it will be shown in section 1.3.10, the fact that it does not conserve the resultant force vector and total energy at the interface can deteriorate its accuracy. In [[Coniglio et al., 2018a](#)] the author proposed a new approach that shares with Internodes its simplicity but achieves an improved accuracy. This one is based on another expression of the continuity of the displacement field at the interface. Before introducing the new formulation let us highlight some equations that can be derived. Each line of the vector coming from the multiplication of the interface mass matrix and the interface displacement vector represents the integral over the interface of the displacement field times the shape function of the DOF corresponding to the selected line. It

is possible to define a weighted average displacement field using as weight the shape function:

$$\bar{u}_{\Gamma_1 i} = \frac{\int_{\Gamma_1} \{u\}(\{x\} \|_{\Gamma_1}) \cdot [N]_i^{\Gamma_1}(\{x\} \|_{\Gamma_1}) d\Gamma_1}{\int_{\Gamma_1} ([N]_i^{\Gamma_1}(\{x\} \|_{\Gamma_1}))_i d\Gamma_1} \quad (1.100)$$

this can be reformulated as:

$$(M_1 u_{\Gamma_1})_i = S_{N_i} \bar{u}_{\Gamma_1 i} \quad (1.101)$$

where  $S_{N_i}$  is the integral of the  $i$ -th shape function over its support and  $\bar{u}_{\Gamma_1 i}$  is the weighted average of the  $i$ -th displacement on  $\Gamma_1$ . One can put this relationship in the matrix form:

$$[M]_1 \{u\}_{\Gamma_1} = [S]_1 \{\bar{u}\}_{\Gamma_1} \quad [M]_2 \{u\}_{\Gamma_2} = [S]_2 \{\bar{u}\}_{\Gamma_2} \quad (1.102)$$

where  $[S]_1$  and  $[S]_2$  denote the diagonal matrices containing on the diagonal the integral of each shape function on its support<sup>10</sup>, and with  $\{\bar{u}\}_{\Gamma_1}$  and  $\{\bar{u}\}_{\Gamma_2}$  the weighted average displacement field of each component over the corresponding shape function support. In the proposed approach the continuity of the weighted average displacement interpolating between the two surfaces is enforced i.e.

$$\{\bar{u}\}_{\Gamma_2} = [\Pi]_{21} \{\bar{u}\}_{\Gamma_1} \quad (1.103)$$

By substitution of equation (1.102) into equation (1.103) one gets a new interpolation operator between the displacement fields:

$$\{u\}_{\Gamma_2} = [M]_2^{-1} [S]_2 [\Pi]_{21} [S]_1^{-1} [M]_1 \{u\}_{\Gamma_1} = [\Pi]_{21}^* \{u\}_{\Gamma_1} \quad (1.104)$$

If the interpolation operator  $[\Pi]_{21}$  satisfies the conservation conditions given by equation (1.60),  $[\Pi]_{21}^*$  will also satisfy the same conditions:

$$\begin{aligned} [\Pi]_{21}^* [1]_{\Gamma_1} &= [M]_2^{-1} [S]_2 [\Pi]_{21} [S]_1^{-1} [M]_1 [1]_{\Gamma_1} = [M]_2^{-1} [S]_2 [\Pi]_{21} [S]_1^{-1} [S]_1 [1]_{\Gamma_1} = \\ &= [M]_2^{-1} [S]_2 [\Pi]_{21} [1]_{\Gamma_1} = [M]_2^{-1} [S]_2 [1]_{\Gamma_2} = [M]_2^{-1} [M]_2 [1]_{\Gamma_2} = [1]_{\Gamma_2} \end{aligned} \quad (1.105)$$

where it was used twice the fact that the mass coherent and lumped mass matrices conserve the sum of lines i.e.

$$[M]_1 [1]_{\Gamma_1} = [S]_1 [1]_{\Gamma_1} \quad [M]_2 [1]_{\Gamma_2} = [S]_2 [1]_{\Gamma_2} \quad (1.106)$$

The condition of zero work at the interface (equation (1.57)) can be used as was done in the classic elimination methods so that the final matrix form of the WACA approach is:

$$\begin{bmatrix} [\mathbf{K}]_{1,1} & [\mathbf{K}]_{1,\Gamma_1} & [0] \\ [\mathbf{K}]_{\Gamma_1,1} & [\mathbf{K}]_{\Gamma_1,\Gamma_1} + ([\Pi]_{21}^*)^T [\mathbf{K}]_{\Gamma_2,\Gamma_2} [\Pi]_{21}^* & ([\Pi]_{21}^*)^T [\mathbf{K}]_{\Gamma_2,2} \\ [0] & [\mathbf{K}]_{2,\Gamma_2} [\Pi]_{21}^* & [\mathbf{K}]_{2,2} \end{bmatrix} \begin{Bmatrix} \{u\}_1 \\ \{u\}_{\Gamma_1} \\ \{u\}_2 \end{Bmatrix} = \begin{Bmatrix} \{f\}_1 \\ \{f\}_{\Gamma_1} + ([\Pi]_{21}^*)^T \{f\}_{\Gamma_2} \\ \{f\}_2 \end{Bmatrix} \quad (1.107)$$

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<sup>10</sup> $[S]_1$  and  $[S]_2$  are also denoted as lumped mass matrices

This formulation shares the same simplicity of the Internodes approach, on the other hand satisfies *a priori* the conservation of the residual sum at the interface and of the energy at the interface.

### 1.3.9 *A priori* conservation of moments

All the methods proposed here do respect the balance of residual resultant and residual work but do not respect an *a priori* condition on the moments of the resultant at the interface. In the work of Puso [Puso, 2004], to enforce this balance of moments and keep the same formulation of the mortar method the nodes of the slave surface in the undeformed configuration are moved on the interface surface. This method is interesting but still not very practical to conserve the meshes of both sub-domains. Another interesting approach is proposed by Park et al. in [Park et al., 2002], where a third surface and mesh (the frame) are introduced and the balance of moments and of the patch test are satisfied choosing the position of nodes on the frame. To the authors' best knowledge no works exist which propose to satisfy the moments' balance equation through an *a priori* condition on the interpolation operators  $\Pi_{12}$ ,  $\Pi_{21}$ . This is the objective of this subsection in which a necessary condition is determined and used to correct the projection operator for all eliminations methods as well as for the new WACA method. First of all let's write the balance of moments of the residual:

$$\sum_{i=1}^{n_{\Gamma_1}} \{OP\}_{i\Gamma_1} \times \{R\}_{i\Gamma_1} + \sum_{j=1}^{n_{\Gamma_2}} \{OP\}_{j\Gamma_2} \times \{R\}_{j\Gamma_2} = \{0\} \quad (1.108)$$

$\{OP\}_{i\Gamma_s}$  is defined as the vector connecting the fixed point O to the positions of the  $i^{th}$  nodes of the  $\Gamma_s$  surface ( $s=1,2$ );  $\{R\}_{i\Gamma_s}$  as the vector of the residual force at the same node. This equation must be satisfied for each possible combination of residuals. For the energy balance (1.59) the residuals are not independent so that all the combinations of residuals can be obtained as all the vectors in  $\{R\}^{N_{\Gamma_2}}$  where  $N_{\Gamma_2} = n_d n_{\Gamma_2}$  is the number of DOFs of the second interface. Satisfying equation (1.108) for all vectors of  $\{R\}^{N_{\Gamma_2}}$  means that it has to be verified for each vector in a normal basis, for example the canonical one. For each node  $j$  and for each direction  $n_d^{(j)}$  (for each  $k_2$  DOF) it is possible to write 3 equations as follows:

$$\sum_{i=1}^{n_{\Gamma_1}} \{OP\}_{i\Gamma_1} \times (\{R\}_{i\Gamma_1})_{n_d^{(j)}} + \{OP\}_{j\Gamma_2} \times (\{R\}_{j\Gamma_2})_{n_d^{(j)}} = \{0\} \quad \forall k_2 \in \{1, 2, \dots, N_{\Gamma_2}\} \quad (1.109)$$

By the use of (1.59) one can replace  $(\{R\}_{i\Gamma_1})_{n_d^{(j)}}$  by the  $(k_2, \{k_1\})$  terms of  $[\Pi]_{21}$ , that will be denoted here as  $(\{\Pi_{21}\})_{(k_2, \{k_1\})}$ , where  $\{k_1\}$  are the indexes of the DOFs of the  $i^{th}$  node of  $\Gamma_1$ . Choosing  $O \equiv P_{j\Gamma_2}$  to write each moment equation, equation (1.109) becomes:

$$\sum_{i=1}^{n_{\Gamma_1}} \{P_{j\Gamma_2} P_{i\Gamma_1}\} \times (\{\Pi_{21}\})_{(k_2, \{k_1\})}^T = \{0\} \quad \forall k_2 \in \{1, 2, \dots, N_{\Gamma_2}\} \quad (1.110)$$

This equation can be also written in the following matrix form:

$$(\{\Pi_{21}\})_{k_2} \cdot [B]_{k_2} = \{0\} \quad \forall k_2 \in \{1, 2, \dots, N_{\Gamma_2}\} \quad (1.111)$$

Where  $(\{\Pi_{21}\})_{k_2}$  is the  $k_2^{th}$  line of  $[\Pi]_{21}$ ,  $[B]_{k_2}$  is a matrix  $N_{\Gamma_1} \times 3$  defined by:

$$([B]_{k_2})_{(k_1, n_d)} = (\{P_{j\Gamma_2} P_{i\Gamma_1}\} \times \{n_d\}^{(k_1)})_{n_d} \quad (1.112)$$

$\{n_d\}^{(k_1)}$  is the  $3 \times 1$  versor of the  $k_1^{th}$  DOF direction and by  $(\bullet)_{n_d}$  the extraction of the  $n_d$  component of a vector. Equation (1.111) and equation (1.60) form a system of 6 equations for each line of the projecting operator that can be compactly written as:

$$(\{\Pi_{21}\})_{k_2} \cdot [A]_{k_2}^T = (\{\Pi_{21}\})_{k_2} \cdot \begin{bmatrix} [1]_{\Gamma_1} & [B]_{k_2} \end{bmatrix} = \begin{bmatrix} (\{1_{\Gamma_2}\})_{k_2} & \{0\} \end{bmatrix} = \{b\}^T \quad (1.113)$$

The projection operator coming from an elimination approach (RBF, WRM, WACA) can be corrected to respect equation (1.113). Since significant modifications of the previous interpolations are not desirable, the new line  $(\{\Pi_{21}\})_{k_2}^{(c)}$  should be as close as possible to the original one  $(\{\Pi_{21}\})_{k_2}$  and should satisfy equation (1.113). In other terms the following optimization problem need to be solved:

$$\left\{ \begin{array}{l} \min_{\{s\}} \left( \{s\} - (\{\Pi_{21}\})_{k_2} \right)^T \cdot \left( \{s\} - (\{\Pi_{21}\})_{k_2} \right) \\ \{s\} \cdot [A]_{k_2}^T = \{b\}^T \end{array} \right. \quad (1.114)$$

Where  $(\{\Pi_{21}\})_{k_2}^{(c)}$  was denoted as  $\{s\}$  for brevity. To solve this constrained optimization problem the Lagrangian approach can be used. The stationary condition of the Lagrangian form a linear system of equations that can be solved to find  $\{s\}^T$  is:

$$\left( (\{\Pi_{21}\})_{k_2}^{(c)} \right)^T = (\{\Pi_{21}\})_{k_2}^T + [A]_{k_2}^T \cdot ([A]_{k_2} \cdot [A]_{k_2}^T)^{-1} (\{b\} - [A]_{k_2} \cdot (\{\Pi_{21}\})_{k_2}^T) \quad (1.115)$$

The resulting operator  $[\Pi]_{21}^{(c)}$  will then respect both the balance of resultant of residual/energy and moments. Still, its precision in terms of displacement and stress continuity could be affected, so in the next section its numerical efficiency will be analyzed. The correction presented here can be adapted even for the Internodes approach. The corrected scheme will modify the column of the  $[Q]_{12}$  and not the line of  $[\Pi]_{21}$ . The resulting scheme will therefore conserve the balance of forces and moments but not of energy since  $[Q]_{12}^{(c)} \neq [\Pi]_{21}^T$ .

### 1.3.10 Benchmarking on numerical test cases

#### 1.3.10.1 Test cases definition and analysis for consistent meshes

In this section two test case geometries are presented, each investigated under pure traction and bending-traction boundary conditions, which will serve for benchmarking various methods considered. The first case is a column like structure with spherical ends see figure 1.17a.

In the second case (figure 1.17b) the bottom structure is wider and shorter than the upper one and both ends are spherical. Both structures are completely fixed at the bottom face and for the pure traction loading they are loaded on the upper surface with a constant surface traction in the z direction of magnitude 30.25 MPa. The analysis made for configuration (1) is actually very similar to the ones frequently studied in the literature (cf. [Song et al., 2017]). The main differences consist in the clamping on the bottom side and in curved instead of planar upper and bottom faces. Hence the proposed test does not have a closed form solution and the fine and consistent mesh was considered as reference for the analysis. One should also keep in mind that none of the presented methods passes linear patch tests for curved interfaces. Anyway classic constant stress patch test is not a necessary condition for optimality convergence [Stummel, 1980], other conditions have to be fulfill like in the Generalized patch test [Stummel, 1979] or in the FEM test [Shi, 1987]. On the other hand one could be wondering how much the error induced by the violation of the patch test can affect a general solution. The following benchmarking will try to bring some insights to this question.

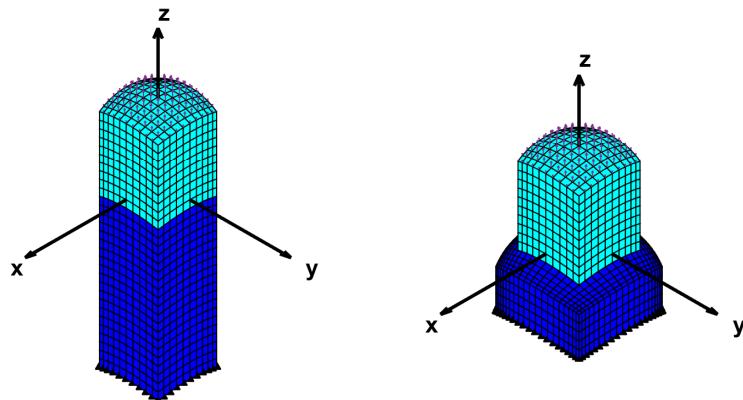
In the FEM, the upper and the lower domains are meshed with linear brick finite elements with complete quadrature (8 Gauss points). In configuration (1) in figure 1.17a the upper and the lower domains are meshed with  $10 \times 10 \times 10$  and  $10 \times 10 \times 20$  elements respectively. Configuration (1) represents a very simple configuration and a more general case in which the interface surfaces do not have the same boundaries was also sought. To achieve this configuration (2) was considered, see figure 1.17b. In this case the nodes that are not inside the boundary of the intersection of the interface surfaces have to be eliminated from the interface node set. To find these nodes one has to check the interpolation operator (applicable for RL-RBF but also for ES) looking for all-zero column and all-zero lines and eliminate the corresponding nodes from the set of interface node. This avoids major errors of projection and gives much better results for all the methods considered. In configuration (2) in figure 1.17b the upper domain is meshed in the same way and the lower domain with  $20 \times 20 \times 5$  elements. The total active DOFs are 10890 in configuration (1) and 10245 in configuration (2). These configurations are denoted in the rest of this paper as reference configurations and will serve for comparison, since they involve consistent meshes between the upper and lower domains and involve the most refined meshes.

In figure 1.18 a bending-traction loading condition is applied to both geometrical configurations. In this case a second surface traction component was added in the x direction with the same magnitude as the one in z direction (30.25MPa). The upper and the lower domains are meshed with consistent meshes, so that this analysis does not introduce any interpolation error between the meshes. The displacement field corresponding to the pure traction and to the bending-traction loadings are represented in figure 1.19 and 1.20 respectively.

The finite element code (in house code, implemented in Matlab) used for this test has been tested and validated with a comparison with Abaqus 6.14. One can observe that in these configurations the displacement field is continuous at the interface between the upper and the lower domain. To represent the Von Mises stress, that is evaluated at each Gauss integration point a global least square interpolation approach [Hinton and Campbell, 1974] was adopted (cf. figure 1.21 and 1.22)<sup>11</sup>. One can observe that for these meshes in both configurations the

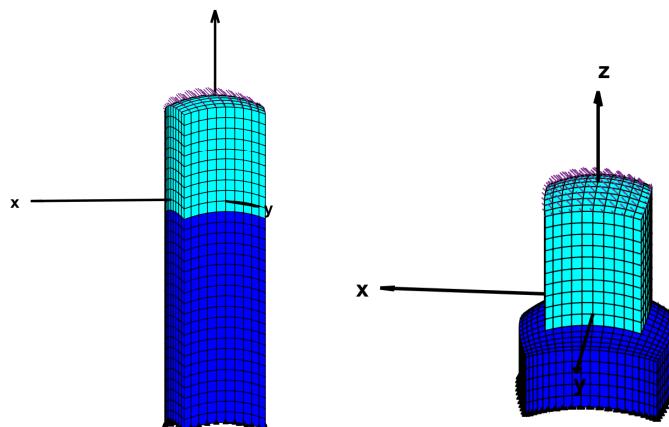
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<sup>11</sup>In such approach the value of von Mises stress is supposed to be interpolated by finite element shape functions. As the values of von Mises stress are computed at the Gauss points, a least square approach is



(a) Geometric configuration (1) (b) Geometric configuration (2)

Figure 1.17: Geometric configurations (1) and (2), pure traction load case



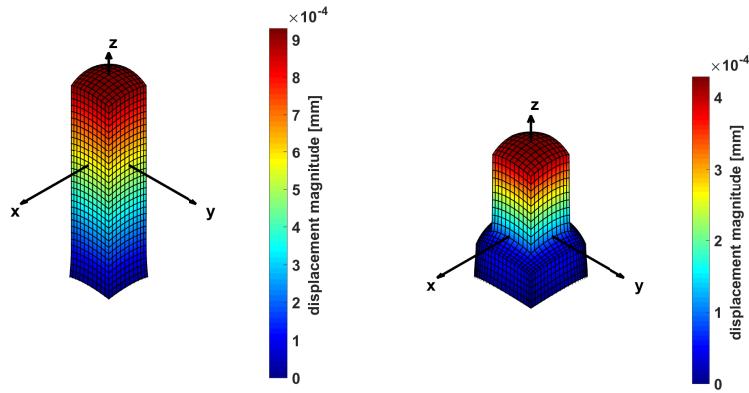
(a) Geometric configuration (1) (b) Geometric configuration (2)

Figure 1.18: Geometric configurations (1) and (2), bending-traction traction load case

Von Mises stress is continuous at the interface.

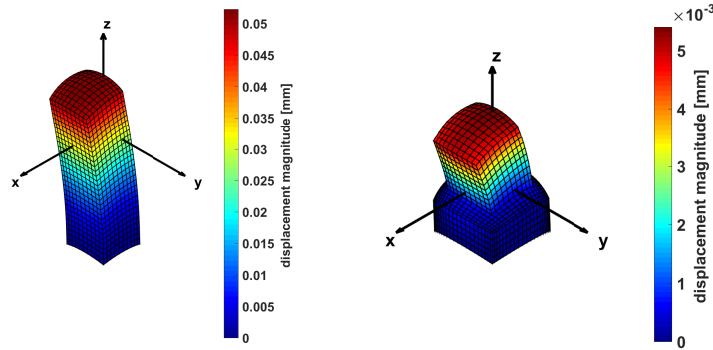
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adopted to determine the nodal values that minimize the difference between the interpolated von Mises stress and the computed ones at Gauss point locations.



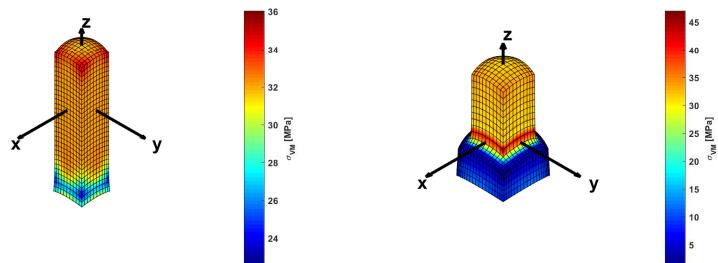
(a) Geometric configuration (1) (b) Geometric configuration (2)

Figure 1.19: Displacement amplitude under pure traction load case



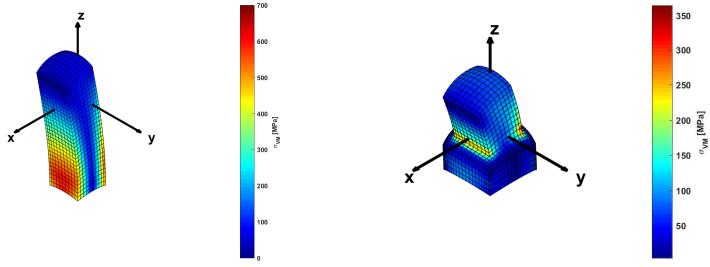
(a) Geometric configuration (1) (b) Geometric configuration (2)

Figure 1.20: Displacement amplitude under bending-traction load case



(a) Geometric configuration (1) (b) Geometric configuration (2)

Figure 1.21: Von Mises stress under pure traction load case



(a) Geometric configuration (1) (b) Geometric configuration (2)

Figure 1.22: Von Mises stress under bending-traction load case

### 1.3.11 Benchmark Results

The inconsistent meshes that are tested here are obtained by changing the mesh of the bottom domain and keeping constant the mesh of the upper domain. The upper domain will always be a  $(10 \times 10 \times 10)$  domain, on the other hand the bottom domain will change its mesh as  $(n \times n \times 20)$  for configuration (1) and as  $(m \times m \times 5)$  in the configuration (2), with  $n$  and  $m$  varying.

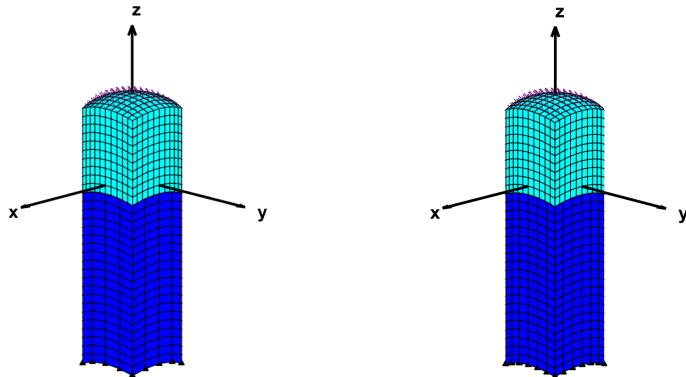
(a) Geometric configuration (1)  
 $n = 4$  (b) Geometric configuration (1)  
 $n = 6$ 

Figure 1.23: Configuration (1) example of inconsistent meshes

One can observe that configuration (1) represents the scenario of figure 1.10. On the other hand in configuration (2) both the inconsistency of figure 1.9 and 1.10 are encountered. To measure the quality of a given mesh tying technique, several parameters are studied as a

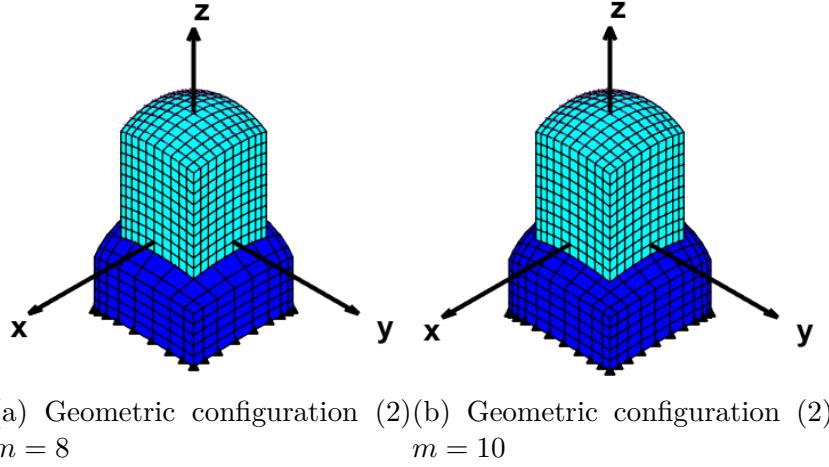


Figure 1.24: Configuration (2) example of inconsistent meshes

function of the discretization of each domain:

- The interface must be in balance of force and moments. The sum of the residuals at one side must be the opposite value of the sum of residuals on the other side. The same for the moments. The percent resultant force and moment relative error is here defined as:

$$E_R = \frac{\|\{R\}_{\Gamma_1} + \{R\}_{\Gamma_2}\|}{\|\{R\}_{\Gamma_1}\|} \times 100\% \quad E_M = \frac{\|\{\mathcal{M}\}_{\Gamma_1} + \{\mathcal{M}\}_{\Gamma_2}\|}{\|\{\mathcal{M}\}_{\Gamma_1}\|} \times 100\% \quad (1.116)$$

Where  $\{R\}_{\Gamma_i}$  indicates the vector of the sum of the residual on the interface  $\Gamma_i$ , and  $\{\mathcal{M}\}_{\Gamma_i}$  indicates the sum of residual moments around a fixed point (in this case the origin).

- In the same way, the total work of the internal forces at the interface should vanish. For this reason one must also consider the compliance error as:

$$E_c = \left| \frac{\{r\}_{\Gamma_1}^T \{u\}_{\Gamma_1} + \{r\}_{\Gamma_2}^T \{u\}_{\Gamma_2}}{\{r\}_{\Gamma_1}^T \{u\}_{\Gamma_1}} \right| \times 100\% \quad (1.117)$$

- The displacement field has to be continuous at the interface. An indicator of displacement discontinuity can be considered as:

$$E_d = \left( \frac{\|\{u\}_{\Gamma_1} - [\Pi]_{12} \{u\}_{\Gamma_2}\|}{2\|\{u\}_{\Gamma_1}\|} + \frac{\|\{u\}_{\Gamma_2} - [\Pi]_{21} \{u\}_{\Gamma_1}\|}{2\|\{u\}_{\Gamma_2}\|} \right) \times 100\% \quad (1.118)$$

This indicator quantifies the amplitude of gaps and com-penetrations at both interfaces' mesh nodes.

- If  $m \leq 20$  and  $n \leq 10$  the conforming meshes can be considered as reference solutions.

One can then compare the displacement field at the node of the constant mesh side (upper domain) with the one in the conforming cases figure 1.17 and 1.18.

$$E_U = \frac{1}{N_{\Gamma_1}} \sum_{i=1}^{N_{\Gamma_1}} \frac{\|\{u\}_{\Gamma_1 i} - \{u\}_{\Gamma_1 i}^{ref}\|}{\|\{u\}_{\Gamma_1 i}^{ref}\|} \times 100\% \quad (1.119)$$

$N_{\Gamma_1}$  indicates the number of nodes of  $\Gamma_1$  (interface surface of the upper domain) and  $\{u\}_{\Gamma_1 i}$  is the displacement vector in the  $i$ -th node of the same surface. The drawback of these indicators is that they are affected by the discretization in each sub-domain as well as they are affected by the interface interpolation.

- The convergence through reference Von Mises stress can also be studied comparing upper domain Gauss point stresses with the corresponding one in reference configuration. Once again the upper domain is unchanged as are its Gauss point locations. The average and the maximum Von Mises stress relative error over the entire upper domain Gauss points are studied.

$$E_S = \frac{1}{N_{PG_1}} \sum_{i=1}^{N_{PG_1}} \frac{|\{\sigma\}_i - \{\sigma\}_i^{ref}|}{|\{\sigma\}_i^{ref}|} \times 100\% \quad (1.120)$$

$$E_\sigma = \max_i \left( \frac{|\{\sigma\}_i - \{\sigma\}_i^{ref}|}{|\{\sigma\}_i^{ref}|} \times 100\% \right) \quad (1.121)$$

Where  $N_{PG_1}$  is the number of Gauss integration points in the upper domain finite elements.

- To complete the comparison the evaluation time will also be considered for each approach.

Different combinations can be adopted using methods described in section 3, For conciseness here only a few methods will be considered:

- Re-Localized Radial Basis Function (RL-RBF) interpolation operator as described in subsection 1.3.5.1
- Weighted residual method (WRM) as described in subsection 1.3.5.3. This approach has the same accuracy of Mortar of subsection 1.3.6.1 but is less expensive in terms of computational effort due to the reduced number of variables. Since the solution of the WRM method has been proven to be the same solution as the Mortar method (cf. [Jeong et al., 2017]) the label "WRM/Mortar" will be employed throughout the benchmark results.
- The Internodes approach as described in subsection 1.3.7 and RL-RBF interpolation operators.
- the Weighted Average Continuity Approach (WACA) following the description of subsection 1.3.8 also using the same RL-RBF interpolation operators.

For the benchmarking of these methods in configuration (1), the mesh refinement  $n$  of the bottom domain is made varying between 4 and 10. The choice of Master and Slave surfaces is also varied and both loading conditions (pure traction and bending-traction) are considered. Furthermore the application or not of the proposed moment correction approach is also considered. The detailed results of these parametric studies are provided in the Appendix. Similarly for configuration (2) the mesh refinement of the bottom domain  $m$  is varied between 8 and 20. The detailed results for configuration (2) are also provided in the Appendix. In order to summarize these parametric studies presented in the Appendix in the main section are provided, box plots that aggregate the error measures for all the different cases considered, cf. Fig. 1.25. In these plots the results are grouped by method and application of moment correction. The labeling "corrected method" means that the proposed moment correction was applied to the respective method. For recall, the line in the middle of the box is the median while the edges of the box represent the 25% and 75% percentiles. The whiskers extend to the most extreme data points not considered outliers, and the outliers are plotted individually by crosses in the plot.

Several conclusions can be drawn from the plots of Fig. 1.25. Apriori moment correction not only makes it possible to satisfy the exact balance of moments as it is clear from figure 1.25b, moreover it sensibly improves the accuracy of all elimination approaches in terms of stress and displacements discrepancy (cf. figures 1.25d, 1.25e, 1.25f and 1.25g). For the computation time on the other hand this projection can cost less than the 25% of the overall CPU time, but improved implementation based on the use of matrices could further reduce this cost.

Overall the most accurate method is found to be WRM/Mortar, which is in accordance with the literature. Unfortunately, as often noted in the literature as well, the accuracy of the WRM/Mortar approach comes at the expense of a significant implementation complexity and significant computational cost (cf. Fig. 1.25h). This is due to the fact that the WRM/Mortar method needs the tedious element to element projections and integrations for the mass matrix assembly that are not necessary in the other approaches. In [De Boer et al., 2007] the WRM method was described as ineffective for fluid structure applications with curved interfaces. Here it is shown that for elastostatic problems this method does not suffer of this weakness. The three other methods (RL-RBF, Internodes and WACA) have much lower implementation complexity and computational cost but come with different tradeoffs with respect to accuracy. The Internodes method does not appear to be able to achieve the vanishing of the work of the internal forces at the interface (cf. Fig. 1.25c), but this does not appear to necessarily badly affect the accuracy of the displacement field, compared to the other methods.

While on the majority of error metrics all methods appear to perform reasonably well, especially after balance of moments correction, one of the most discriminating error metrics is the discrepancy in the Von Mises stresses, where relatively large errors can still be encountered. In order to obtain a better overview of the actual discrepancy in terms of stresses the stress maps corresponding to each approach before and after moment correction are plotted, for  $n = 4$  and when  $\Gamma_1$  (resp.  $\Gamma_2$ ) is chosen as master surface in Fig. 1.26 (resp. in Fig. 1.27). Artificial stress concentrations are generated by the mesh coupling at the interface especially for RL-RBF, Internodes and WACA (even after moment correction). Note however that the *apriori* balance of moments correction sensibly improves the accuracy of the stress fields for all the methods, which performed very poorly without this correction. It is also important

to note that outliers can have totally unacceptable accuracy in terms of stresses. Overall, in terms of accuracy of the stress field, one can note that the corrected WACA approach allows to achieve, on average, a low error on the stresses and a low dispersion from case to case, as well as less extreme outliers, making it a pertinent alternative to the WRM/Mortar approach, while involving a lower implementation complexity and computational cost. Several other additional conclusions can be drawn based on the detailed study of the dependence of the error measures with mesh density (see figures A.1-A.4). In configuration (1)(figure A.1) it is possible to make following observations:

- $n=10$  is not represented for clarity, nevertheless it was verified that all the error indicators were equal to 0 for  $n=10$ .
- In Figure A.1a, and A.1c one can check that RL-RBF,WACA and WRM/Mortar respecting equations (1.59) and (1.60) consequently conserve force resultant and elastic energy at the interface. On the other hand Internodes, that does not respect these equations, shows a discrepancy in terms of reaction sum and residual work at the interface. This discrepancy is severe when the difference between element areas at the interface is important ( $n=4-7$ ) and when the coarser mesh (always  $\Gamma_2$  in this study) is the master. Moreover the discrepancy becomes more severe when the load case is combined (bending-traction load case).
- None of the methods studied here conserve *a priori* the interface total moment(c.f. figure A.1b). WACA and internodes do not conserve the moments for coarse meshes and in the combined load case. WRM/Mortar and RL-RBF perform better and seem to be reasonably accurate even for coarser mesh size. Of course for the pure traction case all the methods perform well in terms of zero moment conservation.
- In figure A.1d the reader can find that RL-RBF generate openings in the deformed configuration when the interface with the finest mesh ( $\Gamma_1$  in this case) is the master. This is a classic problem of node to segment approaches. WACA improves the continuity of the displacement field but is less accurate than internodes and WRM/Mortar especially in the combined load case. All these methods focus on the displacement field.
- In figure A.1e,A.1f and A.1g one can observe displacements and stress convergence at the upper domain<sup>12</sup> to the reference configuration (finest mesh tested with consistent mesh on the interface). It is also possible to acknowledge that WRM/Mortar is quite accurate even for coarse meshes. The other methods are reasonably accurate for finer meshes.
- From figure A.1e, A.1b and A.1g one can also observe that both WACA and internodes lose their precision in bending-traction load case. The non-conservation of moments may be a cause of these weaknesses in agreements with the accuracy improvements obtained after moment correction.

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<sup>12</sup>As mentioned before the error average over interface ( $\Gamma_1$ ) nodes was considered for displacement and the average over all the upper domain Gauss points was considered for stress

- In figure A.1h it is possible to observe that WRM/Mortar is much slower (almost an order of magnitude) than the other methods, and this difference of cost increases with the problem size and with the interface refinement. The particular implementation chosen for this work has not been optimized to reduce the computational cost, anyway in all implementation the tedious element to element projection is the main source of computational effort.
- In figure A.1d for  $n=5$  for WACA, Internodes and RL-RBF, when  $\Gamma_2$  is master  $E_d\%$  is 0 to the machine precision. This is due to the fact that for this mesh all the nodes of  $\Gamma_2$  are superposed to one node of  $\Gamma_1$ . For that reason choosing the coarsest mesh surface ( $\Gamma_2$ ) as master will also imply the point-wise continuity on each node of ( $\Gamma_1$ ). This is not the case for WRM/Mortar as the continuity is imposed in an integral form.

The same study was conducted for configuration (2) and following additional observations can be made:

- In this case as well consistent mesh accuracy ( $m = 20$ ) is not represented for clarity purpose nevertheless it has been checked that all errors converge to 0 in this configuration.
- The accuracy is much better when  $m$  is a multiple of 4 ( $m = 8, 12, 16, 20$ ) i.e. when  $\Gamma_1$  and  $\Gamma_2$  have the same boundaries like in figure 1.24a. For all other configurations, the error is much higher for all the methods studied. The fact that some elements are cut by the interface boundaries like (cf. figure 1.24b) affects the precision of all the studied methods.
- The error in the total moments balance (figure A.2b) is much smaller in this case for all methods except for Internodes where also the energy and the total reaction as indicated in figures A.2a and A.2c is much higher. Not having exactly the same area of element in both meshes is probably the main cause of this error.
- In terms of displacement continuity at the interface (cf. figures A.2d) , the RL-RBF is this time overtaken by both WACA and Internodes. It must be noted that  $E_d\%$  is small when both interfaces have similar displacements even if this is not the "good" one. RL-RBF's poor accuracy results in gaps appearing at the interface in the deformed configuration especially in pure traction load case.
- Looking at the displacement and stress convergence to the reference solution (figures A.2f and A.2g respectively) WRM/Mortar confirms its accuracy even in this case. On the other hand Internodes and RL-RBF show poor convergence especially for  $m \neq 8, 12, 16, 20$  . WACA lies somewhere in between WRM/Mortar and Internodes.
- The CPU time in seconds (figure A.2h) is still much higher for the WRM, for the same reason as in configuration (1).
- As was highlighted for configuration (1), looking at figures A.2b , A.2e and A.2g a method seems to lose its accuracy for the stress prediction when the total moments balance at the interface is not respected.

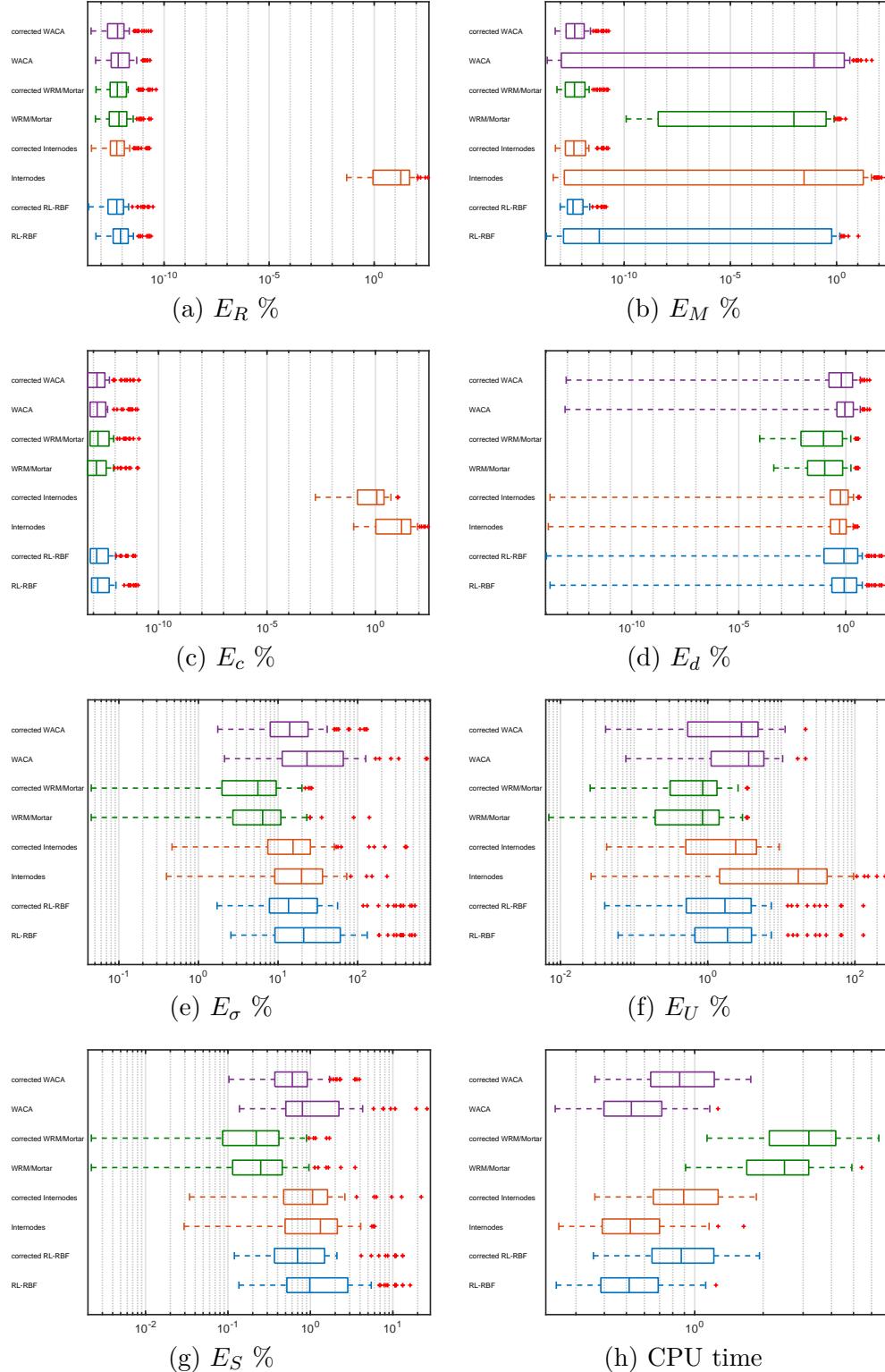


Figure 1.25: Results accuracy and CPU time dispersion over the whole test battery: (a) Boxplot of  $E_R$ , the Resultant Force relative error, (b) Boxplot of  $E_M$ , the moment relative error, (c) Boxplot of  $E_c$ , the interface compliance relative error, (d) Boxplot of  $E_d$ , the displacement discontinuity relative error, (e) Boxplot of  $E_\sigma$ , the maximum of Von Mises stress relative error, (f) Boxplot of  $E_U$ , interface displacement field relative error, (g) Boxplot of  $E_S$ , average Von Mises stress relative error, (h) Boxplot of CPU time (s);

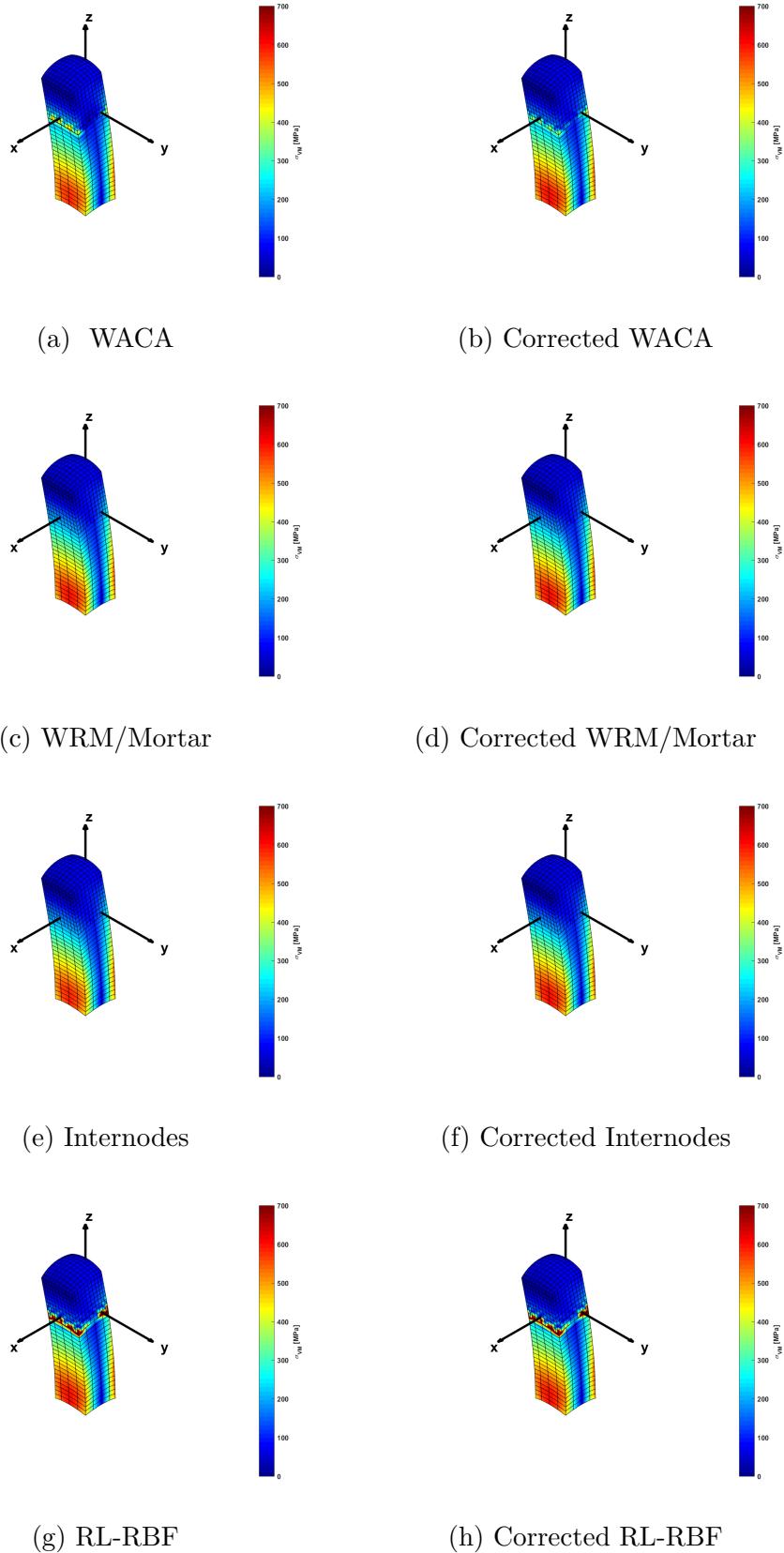


Figure 1.26: Von Mises stress plot configuration (1) under bending-traction load case, for  $n = 4$  and  $\Gamma_1$  master.

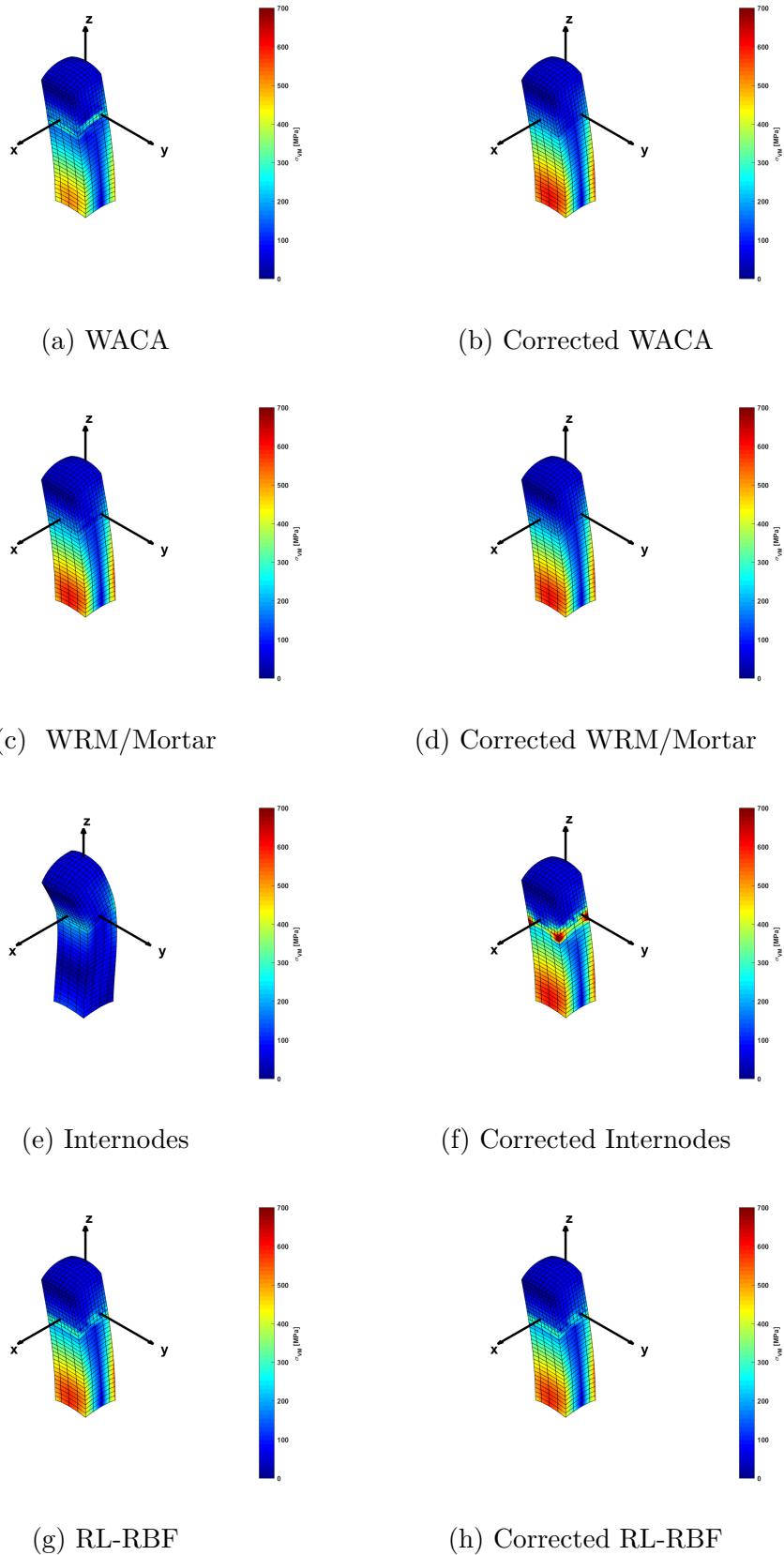


Figure 1.27: Von Mises stress plot configuration (1) under bending-traction load case, for  $n = 4$  and  $\Gamma_2$  master.

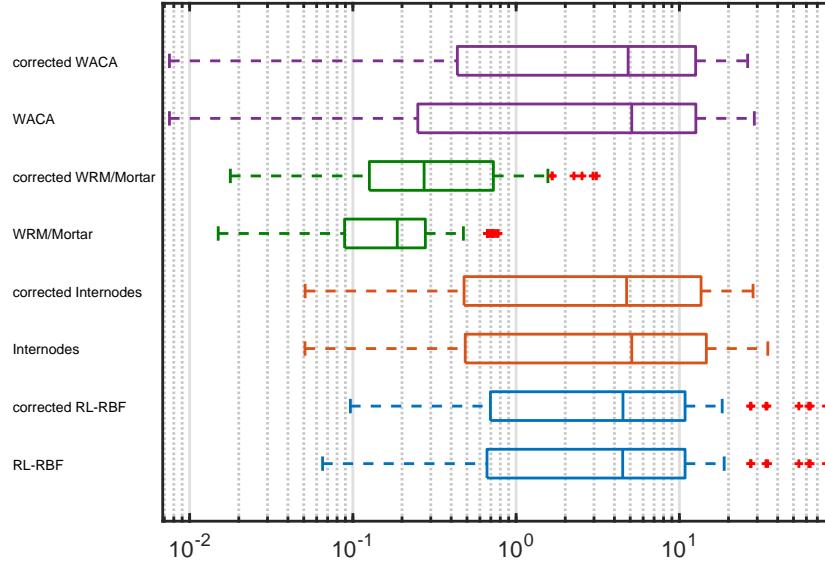


Figure 1.28: Boxplot of  $E_K$ , interface kinetic energy relative error.

There is a last aspect on which a further method could be developed. In this paper the analysis was focused just on elastostatic problem, so it was unnecessary to check for the continuity of the kinetic energy at the interface. Passing to a dynamic problem a relevant error measure is the continuity of the kinematic energy at the interface that can be written as:

$$\{u\}_{\Gamma_1}^T [M]_1 \{u\}_{\Gamma_1} = \{u\}_{\Gamma_2}^T [M]_2 \{u\}_{\Gamma_2} \quad (1.122)$$

A check of this continuity is made here using the kinetic energy error index  $E_k\%$  defined as:

$$E_k = \frac{\{u\}_{\Gamma_1}^T [M]_1 \{u\}_{\Gamma_1} - \{u\}_{\Gamma_2}^T [M]_2 \{u\}_{\Gamma_2}}{\{u\}_{\Gamma_1}^T [M]_1 \{u\}_{\Gamma_1}} \times 100\% \quad (1.123)$$

Figure 1.28 results of all previous experiences are given also for  $E_k\%$ . It is possible to observe from these boxplots that further developments are still possible aimed at improving specifically the accuracy of the methods for dynamic problems.

## 1.4 Advanced solver for FEA

### 1.4.1 Superelement exploitation

In the previous sections a 3D solid finite element formulation was considered. These, even if very general, suffer of strong limitations, when considering degenerated geometry, such as beam or shell geometries. In such structures linear solid elements suffer of some important shortcomings such as shear locking or computational burden. For this reason more specific

component such as beam or shell elements have been developed in many FEM libraries. The Whole Engine Model (WEM) in the context presented in this work, is a very complex assembly of many different kinds of Elements, with many complicated loadings computed from both the engine manufacturer and the load analysis department. In such way of working the WEM needs to be considered with its contributions to loads and stiffness. These two contribute to the final measure of engine displacements that are monitored in this study. The fact that an industrial engine model has to be integrated in the finite element analysis poses a real problem in terms of both implementation efficiency and development time since the combined engine and design zone structural model may be too complex and computationally expensive to be solved at each iteration of the optimization process. To circumvent these issues superelements [Nastran, 2013] were employed. These are very efficient to deal with complex linear models that do not change in the optimization loop [Krog et al., 2004]. Here this method is developed on a general structure shown in Figure 1.29. Sorting the structure's

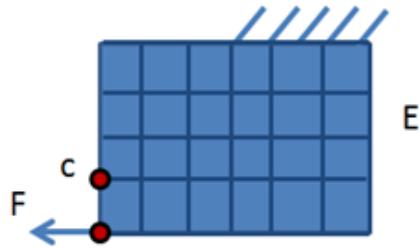


Figure 1.29: Example of DOFs partition: the nodes of the structure are sorted in retained nodes (whose DOFs are ( $c$ )) and other nodes (whose DOFs are ( $E$ )). After the static condensation, a superelement containing only ( $c$ ) DOFs will be generated. Nevertheless the suppressed DOFs ( $E$ ) can still be computed after static analysis thanks to equation 1.125.

degrees of freedom into retained ( $c$ ) and other degrees of freedom ( $E$ )<sup>13</sup>, the static balance equation can then be written in the following form:

$$\begin{bmatrix} [\mathbf{K}_{cc}] & [\mathbf{K}_{cE}] \\ [\mathbf{K}_{Ec}] & [\mathbf{K}_{EE}] \end{bmatrix} \begin{Bmatrix} \{u_c\} \\ \{u_E\} \end{Bmatrix} = \begin{Bmatrix} \{F_c\} \\ \{F_E\} \end{Bmatrix} \quad (1.124)$$

Using the second line block of equations and solving for  $\{u_E\}$

$$\{u_E\} = [\mathbf{P}] \{u_c\} + \{u_0\} \quad (1.125)$$

With

$$[\mathbf{P}] = -[\mathbf{K}_{EE}]^{-1} [\mathbf{K}_{Ec}] \quad (1.126)$$

$$\{u_0\} = [\mathbf{K}_{EE}]^{-1} \{F_E\} \quad (1.127)$$

The  $[\mathbf{P}]$  matrix is called constrained modal matrix in Nastran and recovery matrix in Abaqus.  $\{u_0\}$  is called fixed interface displacement on Nastran and has to be computed in a separated

<sup>13</sup>note that here ( $E$ ) indicates only free DOFs and homogeneous Dirichlet conditions are considered.

analysis on Abaqus. Substituting equation 1.125 back in the first line block of 1.124:

$$[\tilde{\mathbf{K}}_{cc}] \{u_c\} = \{\tilde{F}_c\} \quad (1.128)$$

With

$$[\tilde{\mathbf{K}}_{cc}] = [\mathbf{K}_{cc}] + [P]^T [\mathbf{K}_{Ec}] \quad (1.129)$$

$$\{\tilde{F}_c\} = \{F_c\} + [P]^T \{F_E\} \quad (1.130)$$

For the problem in hand, it is possible to make the evaluation of  $[\tilde{\mathbf{K}}_{cc}]$ ,  $\{\tilde{F}_c\}$ ,  $[P]$  using a commercial software like Abaqus using the engine model. The  $\{u_0\}$  vector can also be evaluated using a linear perturbation static analysis of the engine, fixing the DOFs of the interface. If the structure has to be integrated into another model, the retained DOFs ( $c$ ) can be used to describe the displacements of the whole assembly. The stiffness matrix of equation 1.124 doesn't need to be inverted anymore, the only knowledge of  $[\tilde{\mathbf{K}}_{cc}]$  and  $\{\tilde{F}_c\}$  is sufficient. Considering the engine interface DOFs as retained DOFs ( $c$ ), it is possible to evaluate engine and design zone assembly displacements just considering the retained DOFs stiffness matrix and load vector. Using mesh tying techniques reviewed in section 1.3 the engine retained nodes stiffness matrix and load vector can be integrated to the design zone. In fact let's consider a simple elimination approach for the mesh tying of engine superelement and design zone finite element model.

$$\{u_c\} = [\Pi_{cd}] \{u_d\} \quad (1.131)$$

Where  $\{u_c\}$ ,  $\{u_d\}$  are respectively engine and design zone interface DOFs. After eliminating engine DOFs, the final system of equation reads :

$$\begin{bmatrix} [\mathbf{K}_{oo}] & [\mathbf{K}_{od}] \\ [\mathbf{K}_{do}] & [\mathbf{K}_{dd}] + [\Pi_{cd}]^T [\tilde{\mathbf{K}}_{cc}] [\Pi_{cd}] \end{bmatrix} \begin{Bmatrix} \{u_o\} \\ \{u_d\} \end{Bmatrix} = \begin{Bmatrix} \{F_o\} \\ \{F_d\} + [\Pi_{cd}]^T \{\tilde{F}_c\} \end{Bmatrix} \quad (1.132)$$

Where  $o$  is the index of design zone DOFs not lying on the interface and:

$$[\mathbf{K}_{DZ}] = \begin{bmatrix} [\mathbf{K}_{oo}] & [\mathbf{K}_{od}] \\ [\mathbf{K}_{do}] & [\mathbf{K}_{dd}] \end{bmatrix} \quad (1.133)$$

Is the stiffness matrix relative to the Design Zone of the topology optimization problem and

$$[\mathbf{K}_{Eng}] = \begin{bmatrix} [0] & [0] \\ [0] & [\Pi_{cd}]^T [\tilde{\mathbf{K}}_{cc}] [\Pi_{cd}] \end{bmatrix} \quad (1.134)$$

Is the engine superelement stiffness matrix. The final system of equations in this case can then be written in a short form as :

$$([\mathbf{K}_{Eng}] + [\mathbf{K}_{DZ}]) \{U\} = [\mathbf{K}] \{U\} = \{F\} \quad (1.135)$$

The superelement exploitation implies a significant economy in CPU time especially for big and complex engine models, since the full engine model does not have to be evaluated at each change of the design zone considered by the topology optimization. Furthermore, this also implies that the software used for evaluating the engine model can be separate from the one for evaluating the design zone model and carrying out the topology optimization. This is important since in practice the engine model is usually constructed in commercial FE software. In this work Abaqus was used for the engine model and Matlab for the design zone model and topology optimization. Accordingly reading Abaqus .dat and .mtx file from Matlab environment it is possible to make the evaluation of the whole structure just using the Matlab environment as is summarized in figure 1.30. Abaqus is used to generate both the engine and the design zone. A parsing of the input file is made on Matlab to import the design zone mesh coordinates and finite elements. The engine model is reduced to a superelement on the interface using Abaqus, then a parsing is made on the .mtx and the .dat file coming respectively from the sub-structuring and the linear load case. The stiffness matrix of the design zone is assembled together with the engine superelement stiffness matrix, the same is done for the load vectors. For each iteration of the topology optimization loop, the assembled problem is solved to determine the design zone displacements and the recovered engine displacements. Finally tip clearance variations are computed in each stage and employed to compute consumption variation. This gives significant flexibility in implementing

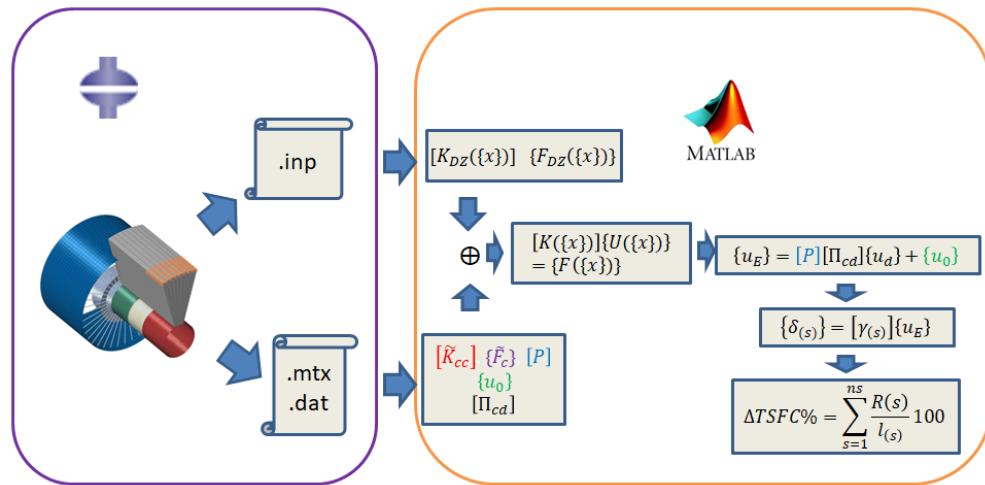


Figure 1.30: Tip-clearance evaluation work-flow.

various approaches developed for topology optimization.

#### 1.4.2 Iterative solvers

Solving the balance equations (1.132) for the free DOFs vector is an expensive task especially for large 3D problem as the one that is tackled in this work. The stiffness matrix is large, sparse and positive definite. For this reason one can either employ direct solvers [Davis, 2006] or iterative solvers [Saad, 2003]. When dealing with small 2D problems, direct solvers are

preferable due to their computational efficiency and robustness. In these approaches, the stiffness matrix is decomposed in the product of a lower triangular and an upper triangular matrices (LU or Choleski decompositions). Then, the resolution two systems of linear equations is carried out as a back substitution operation. For large scale 3D problems, these approaches become too expensive and ineffective [Davis, 2006]. For these cases, the sparsity of the stiffness matrix results in an inexpensive matrix vector product operation which can be effectively utilized by iterative solution algorithms.

#### 1.4.2.1 The Conjugate Gradient Algorithm

For symmetric positive definite matrices, a classic choice in terms of iterative solvers is the Conjugate Gradient (CG) algorithm [Hestenes and Stiefel, 1952]. Here this algorithm is derived following the procedure illustrated in [Saad, 2003]. At each iteration  $k$  of the Conjugate Gradient algorithm, the solution vector  $\{U\}$  is updated as:

$$\{U\}_{(k+1)} = \{U\}_{(k)} + \alpha_{(k)} \{p\}_{(k)} \quad (1.136)$$

where  $\{U\}_{(k+1)}$  is the solution at the next iteration  $k + 1$  that is written in function of the solution at the previous iteration  $\{U\}_{(k)}$ ,  $\alpha_{(k)}$  is a scalar and  $\{p\}_{(k)}$  is the perturbation direction at the same iteration  $k$ . One can define the residual vector  $\{r\}$  defined at each iteration as:

$$\{r\}_{(k)} = \{F\} - [\mathbf{K}] \{U\}_{(k)} \quad (1.137)$$

The residual for the  $k + 1$  iteration can be then written as:

$$\{r\}_{(k+1)} = \{F\} - [\mathbf{K}] \{U\}_{(k)} - \alpha_{(k)} [\mathbf{K}] \{p\}_{(k)} = \{r\}_{(k)} - \alpha_{(k)} [\mathbf{K}] \{p\}_{(k)} \quad (1.138)$$

To enforce the residual at the iteration  $k + 1$  to be orthogonal to the one of the previous iteration  $k$ :

$$\{r\}_{(k+1)}^T \{r\}_{(k)} = 0 \quad (1.139)$$

That is true only if:

$$\alpha_{(k)} = \frac{\{r\}_{(k)}^T \{r\}_{(k)}}{\{r\}_{(k)}^T [\mathbf{K}] \{p\}_{(k)}} \quad (1.140)$$

For the update of the search direction, one can ask that it has to belong to the span of vectors  $\{r\}_{(k+1)}$  and  $\{p\}_{(k)}$ :

$$\{p\}_{(k+1)} = \{r\}_{(k+1)} + \beta_{(k)} \{p\}_{(k)} \quad (1.141)$$

To determine  $\beta_{(k)}$  one has to enforce this time the orthogonality between  $\{p\}_{(k+1)}$  and  $[\mathbf{K}] \{p\}_{(k)}$ :

$$\{p\}_{(k)}^T [\mathbf{K}] \{p\}_{(k+1)} = 0 \quad (1.142)$$

true only if:

$$\beta_{(k)} = -\frac{\{p\}_{(k)}^T [\mathbf{K}] \{r\}_{(k+1)}}{\{p\}_{(k)}^T [\mathbf{K}] \{p\}_{(k)}} \quad (1.143)$$

Being:

$$[\mathbf{K}] \{p\}_{(k)} = -\frac{1}{\alpha_{(k)}} (\{r\}_{(k+1)} - \{r\}_{(k)}) \quad (1.144)$$

So that:

$$\beta_{(k)} = \frac{\{r\}_{(k+1)}^T \{r\}_{(k+1)}}{\alpha_{(k)} \{p\}_{(k)}^T [\mathbf{K}] \{p\}_{(k)}} \quad (1.145)$$

Using equations (1.141) and (1.142):

$$\{p\}_{(k)}^T [\mathbf{K}] \{p\}_{(k)} = \{p\}_{(k)}^T [\mathbf{K}] (\{r\}_{(k)} + \beta_{(k-1)} \{p\}_{(k-1)}) = \{p\}_{(k)}^T [\mathbf{K}] \{r\}_{(k)} \quad (1.146)$$

then:

$$\beta_{(k)} = \frac{\{r\}_{(k+1)}^T \{r\}_{(k+1)}}{\{r\}_{(k)}^T \{r\}_{(k)}} \quad (1.147)$$

The CG algorithm can then be written as:

<b>Input:</b> $[\mathbf{K}], \{F\}, \varepsilon_U, N_{max}$ <b>Output:</b> $\{U\}_{(k)}$ approximation of the solution of $[\mathbf{K}] \{U\} = \{F\}$ <pre> 1  k ← 0; 2  <math>\{U\}_{(0)} \leftarrow \{0\};</math> 3  <math>\{r\}_{(0)} \leftarrow \{F\};</math> 4  <math>\{p\}_{(0)} \leftarrow \{r\}_{(0)};</math> 5  <b>while</b> <math>\frac{\ \{r\}_{(k)}\ _2}{\ \{r\}_{(0)}\ _2} \leq \varepsilon_r</math> or <math>k \geq N_{max}</math> <b>do</b> 6    <math>\alpha_{(k)} \leftarrow \frac{\{r\}_{(k)}^T \{r\}_{(k)}}{\{p\}_{(k)}^T [\mathbf{K}] \{p\}_{(k)}};</math> 7    <math>\{U\}_{(k+1)} \leftarrow \{U\}_{(k)} + \alpha_{(k)} \{p\}_{(k)};</math> 8    <math>\{r\}_{(k+1)} \leftarrow \{r\}_{(k)} - \alpha_{(k)} [\mathbf{K}] \{p\}_{(k)};</math> 9    <math>\beta_{(k)} \leftarrow \frac{\{r\}_{(k+1)}^T \{r\}_{(k+1)}}{\{r\}_{(k)}^T \{r\}_{(k)}};</math> 10   <math>\{p\}_{(k+1)} \leftarrow \{r\}_{(k+1)} + \beta_{(k)} \{p\}_{(k)};</math> 11   <math>k \leftarrow k + 1;</math> 12 <b>end</b> 13 <b>return</b> <math>\{U\}_{(k)}</math></pre>
---

**Algorithm 1:** Conjugate Gradient

### 1.4.2.2 The Preconditioned Conjugate Gradient (PCG) Algorithm

It can be proven [Saad, 2003] that CG reaches the solution vector  $\{U^*\}$  in at most  $n$  iterations<sup>14</sup>, and the convergence rate is shown to be:

$$\|\{U\}_{(k)} - \{U^*\}\|_{[\mathbf{K}]} \leq 2 \left( \frac{\sqrt{\kappa} - 1}{\sqrt{\kappa} + 1} \right)^k \|\{U\}_0 - \{U^*\}\|_{[\mathbf{K}]} \quad (1.148)$$

where the  $[\mathbf{K}]$ -norm of a vector is defined as:

$$\|\{x\}\|_{[\mathbf{K}]} = \sqrt{\{x\} [\mathbf{K}] \{x\}} \quad (1.149)$$

$k$  is the iteration where the CG algorithm was stopped and  $\kappa$  is the condition number of the stiffness matrix  $[\mathbf{K}]$ . Equation (1.148) shows that for  $\kappa \gg 1$  the convergence of CG algorithm is very slow. In practice to accelerate and ensure convergence, preconditioning techniques are employed. The basic idea of these techniques is to replace the original system of equations by another one that has the same solution and is easier to be solved by iterative algorithms. An operator  $[\mathbf{M}]^{-1}$  has to be constructed so that  $\kappa([\mathbf{M}]^{-1} [\mathbf{K}]) \ll \kappa([\mathbf{K}])$ . By the use of such techniques the CG algorithm can be modified as in algorithm 2. One can observe that the

<b>Input:</b> $[\mathbf{K}], [\mathbf{M}]^{-1}, \{F\}, \varepsilon_U, N_{max}$ <b>Output:</b> $\{U\}_{(k)}$ approximation of the solution of $[\mathbf{K}] \{U\} = \{F\}$ <pre> 1 <math>k \leftarrow 0;</math> 2 <math>\{U\}_{(0)} \leftarrow \{0\};</math> 3 <math>\{r\}_{(0)} \leftarrow \{F\};</math> 4 <math>\{z\}_{(0)} \leftarrow [\mathbf{M}]^{-1} \{r\}_{(0)};</math> 5 <math>\{p\}_{(0)} \leftarrow \{z\}_{(0)};</math> 6 <b>while</b> <math>\frac{\ \{r\}_{(k)}\ _2}{\ \{r\}_{(0)}\ _2} \leq \varepsilon_r</math> or <math>k \geq N_{max}</math> <b>do</b> 7   <math>\alpha_{(k)} \leftarrow \frac{\{r\}_{(k)}^T \{z\}_{(k)}}{\{p\}_{(k)}^T [\mathbf{K}] \{p\}_{(k)}};</math> 8   <math>\{U\}_{(k+1)} \leftarrow \{U\}_{(k)} + \alpha_{(k)} \{p\}_{(k)};</math> 9   <math>\{r\}_{(k+1)} \leftarrow \{r\}_{(k)} - \alpha_{(k)} [\mathbf{K}] \{p\}_{(k)};</math> 10  <math>\{z\}_{(k+1)} \leftarrow [\mathbf{M}]^{-1} \{r\}_{(k+1)};</math> 11  <math>\beta_{(k)} \leftarrow \frac{\{z\}_{(k+1)}^T \{r\}_{(k+1)}}{\{z\}_{(k)}^T \{r\}_{(k)}};</math> 12  <math>\{p\}_{(k+1)} \leftarrow \{z\}_{(k+1)} + \beta_{(k)} \{p\}_{(k)};</math> 13  <math>k \leftarrow k + 1;</math> 14 <b>end</b> 15 <b>return</b> <math>\{U\}_{(k)}</math></pre>
---

**Algorithm 2:** Preconditioned Conjugate Gradient

application of the projection operator in 2, is needed once per iteration. For this reason a

<sup>14</sup> $n$  being the number of free DOFs i.e. the length of  $\{F\}$ .

good preconditioning operator should be:

- Cheap: its application should not involve too many operations
- Effective:  $\kappa([\mathbf{M}]^{-1} [\mathbf{K}])$  should be as close as possible to 1.
- Scalable:  $\kappa([\mathbf{M}]^{-1} [\mathbf{K}])$  and its computational burden should be proportional to the problem size  $n$ .
- Robust:  $[\mathbf{M}]^{-1}$  efficiency should not be problem dependent.
- Simple: complexity should be justified by the computational gain.

Among most common general purpose techniques one can find incomplete Cholesky Factorization [Kershaw, 1978], the diagonal compensation [Jacobi, 1845] and the Factorized Sparse Approximate Inverses (FSAI) [Buleev, 1960a, Buleev, 1960b, Meijerink and Van Der Vorst, 1977, Il'in Valery, 1992, Varga, 1960]. All these techniques are well adapted when the matrix of coefficients is not linked with a particular physics. In the next subsection multi-grid approaches are described. These take advantage from the fact that the matrix of coefficients comes from a PDE problem discretization.

#### 1.4.2.3 Multigrid Methods

The convergence of PCG for solving linear systems arising from PDE slows down with the size of the problem. Multigrid methods [Trottenberg et al., 2001] have been initially designed specifically to solve such problems. On the other hands these approaches' success is strongly dependent on the problem at hand and they need attention to get improved performances. They consist in the use of smoothers and coarse level corrections. Considering the residual decomposition over the basis of eigenvectors of the stiffness matrix, it can be shown that residual components associated with higher frequencies, are damped within few iterations by simple smoothing techniques like damped Jacobi, Gauss Seidel or Red-Black Gauss Seidel [Saad, 2003]. On the other hand the low frequency components survive the smoothing step and need to be dealt with by coarse mesh corrections. The proofs of this results can be found in [Saad, 2003], for uniform 1D and 2D discretizations. In the present work non uniform meshes were considered, and this can be challenging for theoretical reasons, since one cannot directly transpose the considerations valid for uniform meshes to non-uniforms ones, and for practical reasons, since prolongation and interpolation operators may not be easily available. For this reason in the literature one can find Algebraic Multigrid (AMG) [Stüben, 2001] that tries to emulate Geometric Multigrid (GMG) performance on the only knowledge of the coefficient matrix. This work focus on hierarchical non uniform hexahedral meshes. This means that fine meshes were considered to be a consequence of a coarse mesh successive refinements. Each mesh refinement is considered as a level in the GMG. In this way it is relatively simple to construct Interpolation and Prolongation operators to transfer vectors through levels. Given a finite element of the coarse mesh, its refinement can be obtained cutting each finite element in 8 as shown in figures 1.31,1.32. For each element of the coarse

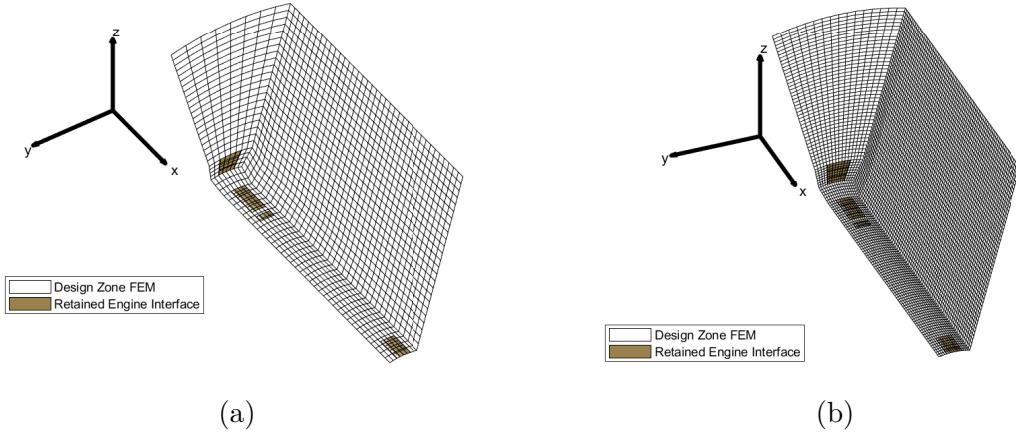


Figure 1.31: Mesh refinement procedure needed for multi-grid approach. The engine reduced element set is colored in yellow. Each node of these elements is kept in the engine superelement.(a) Original design zone mesh, generated in Abaqus and imported on matlab by input file parsing. The mesh counts 7600 8-node linear 3D solid finite elements and 9126 nodes. (b) Design zone mesh after refinement. Each element of the original element is cut in 8 new elements, that give a total of  $7600 \times 8 = 60800$  8-node linear 3D solid finite elements and 66759 nodes.

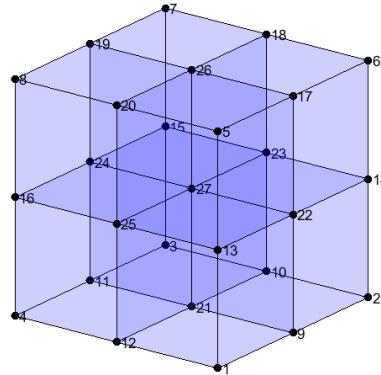


Figure 1.32: 3D partition employed for mesh refinement, nodes 1 to 8 belong to original coarse mesh element. Partition determines node 9 to 27 and 8 elements of the finer mesh.

mesh one can define a transfer of the information between coarse level filed  $u_i^c$   $i = 1, 2, \dots, 8$  and the fine level field  $u_j^f$   $j = 1, 2, \dots, 27$ , that can be used to construct the interpolation operator.

- Node 1 to 8 of the fine mesh have their corresponding node in the coarse level so that:  $u_i^c = u_i^f$  for  $i = 1, 2, \dots, 8$ .

- Nodes 9 to 20 are in the middle of a coarse element edge. For this reason their field can be interpolated using the extremities of edges. For example  $u_9^f = \frac{u_1^c + u_2^c}{2}$ .
- Nodes 21 to 26 correspond to coarse element facet middle points. The corresponding component can be interpolated by the vertex of the corresponding coarse element facets. For instance  $u_{21}^f = \frac{u_1^c + u_2^c + u_5^c + u_6^c}{4}$ .
- Finally node 27 is in the coarse element centroid. Its nodal variable can be interpolated using all the coarse mesh node variables:  $u_{27}^f = \frac{u_1^c + u_2^c + u_3^c + u_4^c + u_5^c + u_6^c + u_7^c + u_8^c}{8}$ .

All previous relationships can be written in the matrix form:

$$\{u_f^{el}\} = [I_{fc}^{el}] \{u_c^{el}\} \quad (1.150)$$

When considering the communication of several independent fields between each level one can use the same interpolation operator for each field:

$$\{U_f^{el}\} = [I_{fc}^{el} \otimes I_d] \{U_c^{el}\} \quad (1.151)$$

where  $\otimes$  denotes the Kronecker product. Finally one can combine the equations coming from each coarse level element to build the global Interpolation operator. In doing so one has to consider each DOF of the fine mesh only once. Generally speaking the interpolation between the  $l^{th}$  refinement and the corresponding coarser level is given by:

$$\{U_l\} = [I_{l,l-1}] \{U_{l-1}\} \quad (1.152)$$

The prolongation operator allows for the inverse communication.

$$\{U_{l-1}\} = [P_{l-1,l}] \{U_l\} \quad (1.153)$$

To keep the stiffness matrix at each refinement symmetric, the prolongation operator is considered to be:

$$[P_{l-1,l}] = [I_{l,l-1}]^T \quad (1.154)$$

In this way given the state problem at the  $l$ -th refinement:

$$[\mathbf{K}_l] \{U_l\} = \{F_l\} \quad (1.155)$$

one can get a state problem at the coarser level as:

$$[I_{l,l-1}]^T [\mathbf{K}_l] [I_{l,l-1}] \{U_{l-1}\} = [\mathbf{K}_{l-1}] \{U_{l-1}\} = [I_{l,l-1}]^T \{F_l\} = \{F_{l-1}\} \quad (1.156)$$

This notation will be used to define the coarse mesh correction step needed in the multigrid. The other important ingredient plays a fundamental role in multigrid convergence is the smoothing. Here the following notation will be used:

$$\{U_{l,v}\} = \text{smooth}^v([\mathbf{K}_l], \{U_{l,0}\}, \{F_l\}) \quad (1.157)$$

where  $\text{smooth}^v$  denotes the application of  $v$  steps of the form:

$$\{U_{l,j+1}\} = [\mathbf{B}_l] \{U_{l,j}\} + [\mathbf{C}_l](\{F_l\}) \quad (1.158)$$

Where the matrix  $[\mathbf{B}_l]$  and  $[\mathbf{C}_l]$  depend on the particular approach chosen for the smoothing and will be detailed later. For two levels the multigrid algorithm can be encoded as in algorithm 3. Note that using this technique recursively, the algorithm moves to coarsen levels

- 1 Pre-smooth:  $\{U_l\} \leftarrow \text{smooth}^{v_1}([\mathbf{K}_l], \{U_l\}, \{F_l\});$
- 2 Get residual:  $\{r_l\} \leftarrow \{F_l\} - [\mathbf{K}_l] \{U_l\};$
- 3 Coarsen:  $\{r_{l-1}\} \leftarrow [I_{l,l-1}]^T \{r_l\};$
- 4 Solve:  $[\mathbf{K}_{l-1}] \{\delta_{l-1}\} = \{r_{l-1}\};$
- 5 Correct:  $\{U_l\} \leftarrow \{U_l\} + [I_{l,l-1}] \{\delta_{l-1}\};$
- 6 Post-smooth:  $\{U_l\} \leftarrow \text{smooth}^{v_2}([\mathbf{K}_l], \{U_l\}, \{F_l\});$

**Algorithm 3:** Two Grid cycle

until the coarsest mesh is reached and direct solvers can indeed be employed to solve the corresponding state problem. Such idea is at the origin of the V-cycle (c.f. algorithm 4). One can observe that in the V-cycle only matrix times vector multiplications are introduced

- 1 Pre-smooth:  $\{U_l\} \leftarrow \text{smooth}^{v_1}([\mathbf{K}_l], \{U_l\}, \{F_l\});$
- 2 Get residual:  $\{r_l\} \leftarrow \{F_l\} - [\mathbf{K}_l] \{U_l\};$
- 3 Coarsen:  $\{r_{l-1}\} \leftarrow [I_{l,l-1}]^T \{r_l\};$
- 4 If  $l = 1;$
- 5 Solve:  $[\mathbf{K}_{l-1}] \{\delta_{l-1}\} = \{r_{l-1}\};$
- 6 Else ;
- 7 Recursion:  $\{\delta_{l-1}\} \leftarrow \text{V-cycle}([\mathbf{K}_{l-1}], \{0\}, \{r_{l-1}\});$
- 8 End ;
- 9 Correct:  $\{U_l\} \leftarrow \{U_l\} + [I_{l,l-1}] \{\delta_{l-1}\};$
- 10 Post-smooth:  $\{U_l\} \leftarrow \text{smooth}^{v_2}([\mathbf{K}_l], \{U_{l,0}\}, \{F_l\});$

**Algorithm 4:**  $\{U_l\} = \text{V-cycle}([\mathbf{K}_l], \{U_l\}, \{F_l\})$

at each prolongation and interpolation step. This operator can be introduced in line 10 of PCG 2. The resulting algorithm also called Multigrid Conjugate Gradients) (MGCG) [Tatebe and Oyanagi, 1994, Ashby and Falgout, 1996] depending on the smoother and the number of levels can be both robust and efficient for solving the linear system associated with the displacement evaluation in topology optimization [Amir et al., 2014, Aage et al., 2017]. As aforementioned, the smoother choice has to be complementary to the coarse mesh corrections in terms of residual error components. In the problem at hand, superelements and non-uniform meshes make this choice particularly important. Damped Jacobi iterations (c.f. 5) are inexpensive and effective options especially for large problems.

- 1  $\{U_l\} \leftarrow \{U_l\} + \omega [\mathbf{D}_l]^{-1} (\{F_l\} - [\mathbf{K}_l] \{U_l\});$

**Algorithm 5:** Damped Jacobi smoother

Where  $[\mathbf{D}_l]$  is the diagonal matrix of  $[\mathbf{K}_l]$ . On the other hand bad mesh quality and static condensation can cause large positive out-of-diagonal terms in the final stiffness matrix that can make diagonal scaling ineffective. Another issue consists in the choice of the damping parameter  $\omega$ . Theoretical considerations can be applied for the uniform meshes as explained in [Saad, 2003]. For non-uniform meshes and for superelements, the accurate choice of this parameter can be even more complicated [Bin Zubair et al., 2010]. One can cite here other effective smoother techniques that can be employed instead of damped Jacobi iterations. The multi-parameter Jacobi method that can be considered as the sequential application of Jacobi step with different values (c.f. algorithm 6). Using such approach, the frequency range

- 1 For  $j = 1, 2, \dots, p$ ;
- 2  $\{U_l\} \leftarrow \{U_l\} + \omega_j [\mathbf{D}_l]^{-1} (\{F_l\} - [\mathbf{K}_l] \{U_l\})$ ;
- 3 End;

**Algorithm 6:** Multi-parameter Jacobi Method

on which damping is effective can be amplified [Bin Zubair, 2009]. One can also consider Gauss-Seidel Iterations. Given the decomposition:

$$[\mathbf{K}_l] = [\mathbf{D}_l] + [\mathbf{L}_l] + [\mathbf{L}_l]^T \quad (1.159)$$

Where  $[\mathbf{L}_l]$  is the lower triangular part of  $[\mathbf{K}_l]$ , then the Gauss-Seidel Iterations takes the form in algorithm 7. For high dimensional problems the effectiveness of these approaches can

- 1  $\{U_l\} \leftarrow ([\mathbf{D}_l] + [\mathbf{L}_l])^{-1} (\{F_l\} - [\mathbf{L}_l]^T \{U_l\})$ ;

**Algorithm 7:** Gauss-Seidel smoother

be improved introducing a relaxation parameter  $0 < \omega < 2$  that gives the Successive Over Relaxation (SOR)method (algorithm 8). Multi-parameter Jacobi principle can be extended

- 1  $\{U_l\} \leftarrow ([\mathbf{D}_l] + \omega [\mathbf{L}_l])^{-1} (\omega \{F_l\} + ((1 - \omega) [\mathbf{D}_l] - \omega [\mathbf{L}_l]^T) \{U_l\})$ ;

**Algorithm 8:** SOR smoother

to SOR approach. In this work this technique is referred to as Multi Parameter Successive Over Relaxation (MPSOR) (algorithm 9).

```

1 For  $j = 1, 2, \dots, p$ ;
2  $\{U_l\} \leftarrow ([\mathbf{D}_l] + \omega_j [\mathbf{L}_l])^{-1} \left( \omega_j \{F_l\} + ((1 - \omega_j) [\mathbf{D}_l] - \omega_j [\mathbf{L}_l]^T) \{U_l\} \right)$ ;
3 EndFor;

```

**Algorithm 9:** MPSOR smoother

For some problems all reviewed approaches may be not efficient, so that Incomplete decomposition smoothers (Ichol(0), ILU(0)) have to be considered. In this case, the stiffness matrix being symmetric:

$$[\mathbf{K}_l] \approx [\bar{\mathbf{L}}_l] [\bar{\mathbf{L}}_l]^T \quad (1.160)$$

where  $[\bar{\mathbf{L}}_l]$  is the incomplete Cholesky decomposition of the stiffness matrix. The ICHOL smoother step is presented in algorithm 10. Smoothers like ICHOL and damped Jacobi can

```

1  $\{U_l\} \leftarrow \{U_l\} + \omega [\bar{\mathbf{L}}_l]^{-1} [\bar{\mathbf{L}}_l]^{-T} (\{F_l\} - [\mathbf{K}_l] \{U_l\})$ ;

```

**Algorithm 10:** ICHOL smoother

also be written as:

$$\{U_l\}_{k+1} = \{U_l\}_k + \omega_k \{\delta_l\}_{k+1} \quad (1.161)$$

Common choices of damping parameter  $\omega$  are based on experience or theoretical principles for Damped Jacobi, in order to be complementary with coarse mesh correction. In topology optimization this can represent a challenge since each configuration has a different stiffness matrix and potentially several optimal  $\omega$ . To make a reasonable choice of the damping parameter here it is considered that smoother efficiency could be related to the residual 2-norm after the smoothing step. If this is looked as part of a deepest descent algorithm, then the  $\omega$  value can be obtained as the result of a line-search minimization of the residual 2-norm over the descent direction.

```

1 Compute  $\{\delta_l\}$  and  $\{\hat{\delta}_l\} = [\mathbf{K}_l] \{\delta_l\}$ ;
2  $\{r_l\} \leftarrow \{F_l\} - [\mathbf{K}_l] \{U_l\}$ ;
3  $\omega \leftarrow \frac{\{r_l\}^T \{\hat{\delta}_l\}}{\{\hat{\delta}_l\}^T \{\hat{\delta}_l\}}$ ;
4  $\{U_l\} \leftarrow \{U_l\} + \omega \{\delta_l\}$ ;

```

**Algorithm 11:** Line Search  $\omega$  update

This method is not theoretically proven to be more effective than constant  $\omega$  counterpart but can provide results that are comparable with an optimal constant choice of  $\omega$ . This comes at the cost of a more expensive computation of the smoother since a matrix-vector and 2 scalar products have to be computed to use this scheme. When the algorithm 11 is adopted together with a reviewed smoother LS prefix will be employed. The most suitable smoother should then be selected on the base of tests made on a given configuration.

#### 1.4.2.4 Multigrid tuning and smoothers benchmark

As it was stated in the previous subsection, multigrid efficiency comes with the loss of solver robustness. In fact smoother tuning and selection is a problem dependent choice. In this section several combinations of coarse mesh corrections and of smoothers are tested to get faster convergence on the finite element model presented in chapters 2 and 3 for three load cases. The design region's Young Modulus is the one of the full material (Iron) for each finite element. Damped Jacobi, Ichol, SOR, LS-Jacobi and LS Ichol smoothers are implemented. The same number of pre and post smoothing were considered:  $v_1 = v_2 = v$ . Only 1 or two sweeping per iteration were applied. For Line Search variants the value of  $\omega$  was computed for each sweeping step on the other hand for simple variants the following ranges were investigated

$$\omega_{Jacobi} \in [0.2, 0.5] \quad (1.162)$$

$$\omega_{SOR} \in [0.4, 1.6] \quad (1.163)$$

$$\omega_{Ichol} \in [0.9, 1.2] \quad (1.164)$$

The convergence criterion was based on the maximum of the  $L^2$ -norm of each load case's residual and on the maximum number of iterations:

$$r_r = \frac{\max_i \|\{r_{IT,i}\}\|_2}{\max_i \|\{r_{0,i}\}\|_2} \leq 10^{-5} \quad (1.165)$$

$$IT \leq 10^2 \quad (1.166)$$

where  $r_r$  is referred to as relative residual. The 3 load cases were solved simultaneously adapting the original Matlab code of [Amir et al., 2014]. The results are compared in terms of Iterations at convergence, CPU total time  $T_{tot}$ , the average residual reduction per iteration and convergence speed. The average residual reduction per iteration indicates the coefficient that reduces residual norm at each iteration:

$$ARR = \left( \frac{\max_i \|\{r_{0,i}\}\|_2}{\max_i \|\{r_{IT,i}\}\|_2} \right)^{\frac{1}{IT}} \quad (1.167)$$

where  $\{r_{0,i}\}$  indicates the residual of load case  $i$  at iteration 0, and  $\{r_{IT,i}\}$  indicates the residual of load case  $i$  at the last iteration  $IT$ . One can say that at each iteration the residual norm in average is reduced by  $ARR$  times. That is still an incomplete piece of information because the evaluation time per iteration is not considered. Convergence speed (measured in Hertz or reductions per second) can be defined as:

$$C_S = -\frac{IT \log(ARR)}{T_{it}} \quad (1.168)$$

where  $T_{it}$  is the CPU time in seconds of the iterative approach that grows with the PCG iterations. In fact the time needed for the incomplete and full Cholesky decomposition of stiffness matrices in each level has to be conducted just once before PCG iterations. The  $C_S$  parameter is considered the average convergence ratio over the whole convergence. If the convergence history had an exponential form with respect to the computational time from

the beginning of iteration  $t$  it would have the form of:

$$\max_i \| \{r_{IT,i}\} \|_2(t) = \max_i \| \{r_{0,i}\} \|_2 e^{-C_S t} \quad (1.169)$$

The higher the  $C_S$ , the faster is the convergence for the same accuracy. Finally the total CPU time here mentioned as  $T_{tot}$  include both the part coming from iterations and the part coming from decompositions. In figure 1.33 the performances of damped Jacobi smoother are

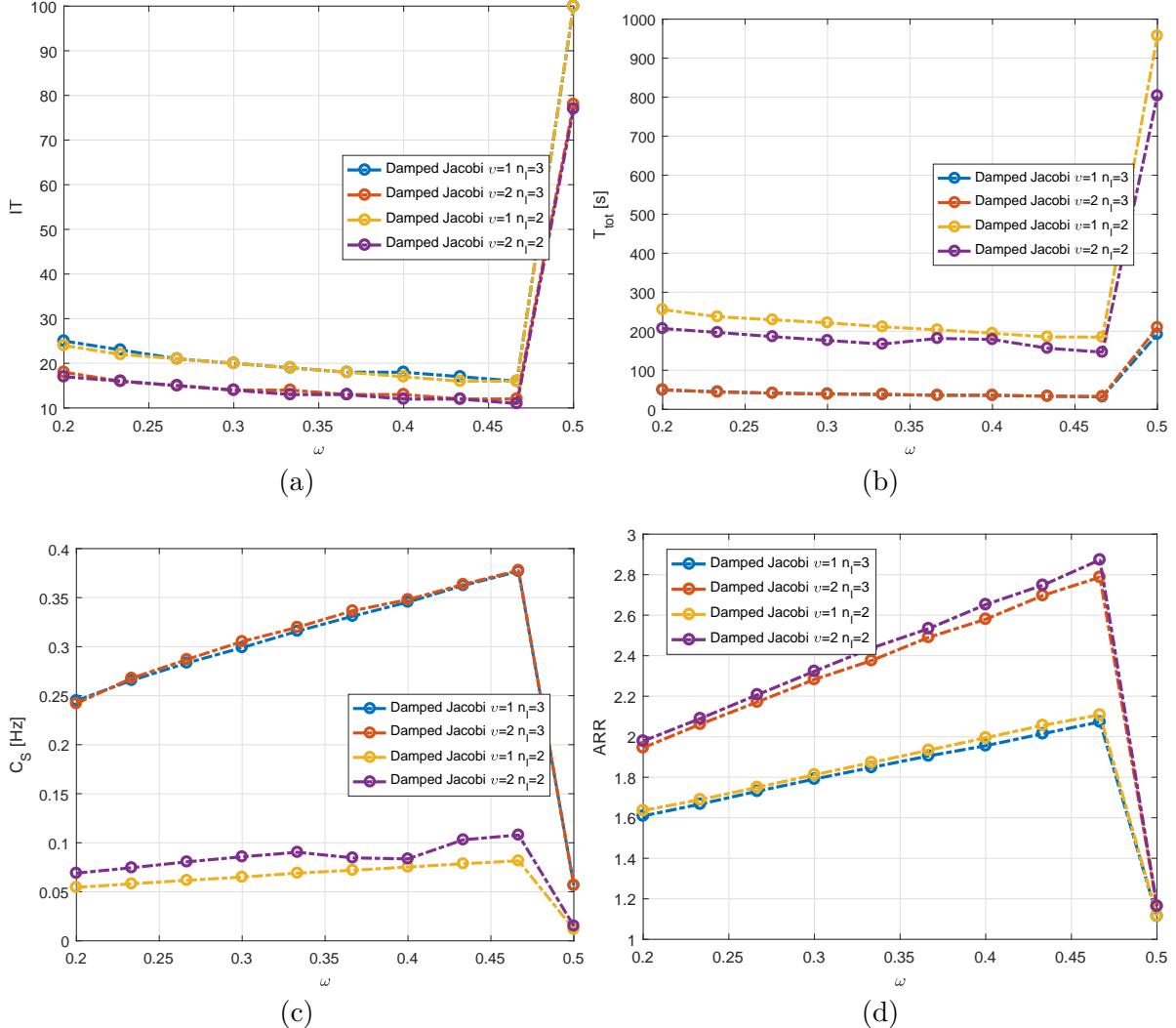


Figure 1.33: Damped Jacobi smoother performances as a function of  $\omega$  and for different numbers of mesh levels  $n_l$ , different numbers of sweeping  $v$ . (a) Iterations at convergence.(b) CPU time [s]. (c) Convergence speed defined in equation [Hz] (1.168). (d) Average Residual Reduction per iteration defined in equation (1.167).

computed for several values of  $\omega$ , mesh levels  $n_l$  and different numbers of sweeping  $v$ . One can observe that the minimal CPU time  $\approx 33$  seconds is reached for  $n_l = 3$ ,  $v = 2$ ,  $\omega \approx 0.47$ . One can observe that for  $\omega = 0.5$  and  $v = 1$  the desired accuracy for the relative residual was not reached within 100 iterations. The maximal convergence speed was of  $0.35\text{Hz}$ , that makes

the error reduced by 2.7 at each iteration ( c.f. figure 1.33d). SOR smoother performances

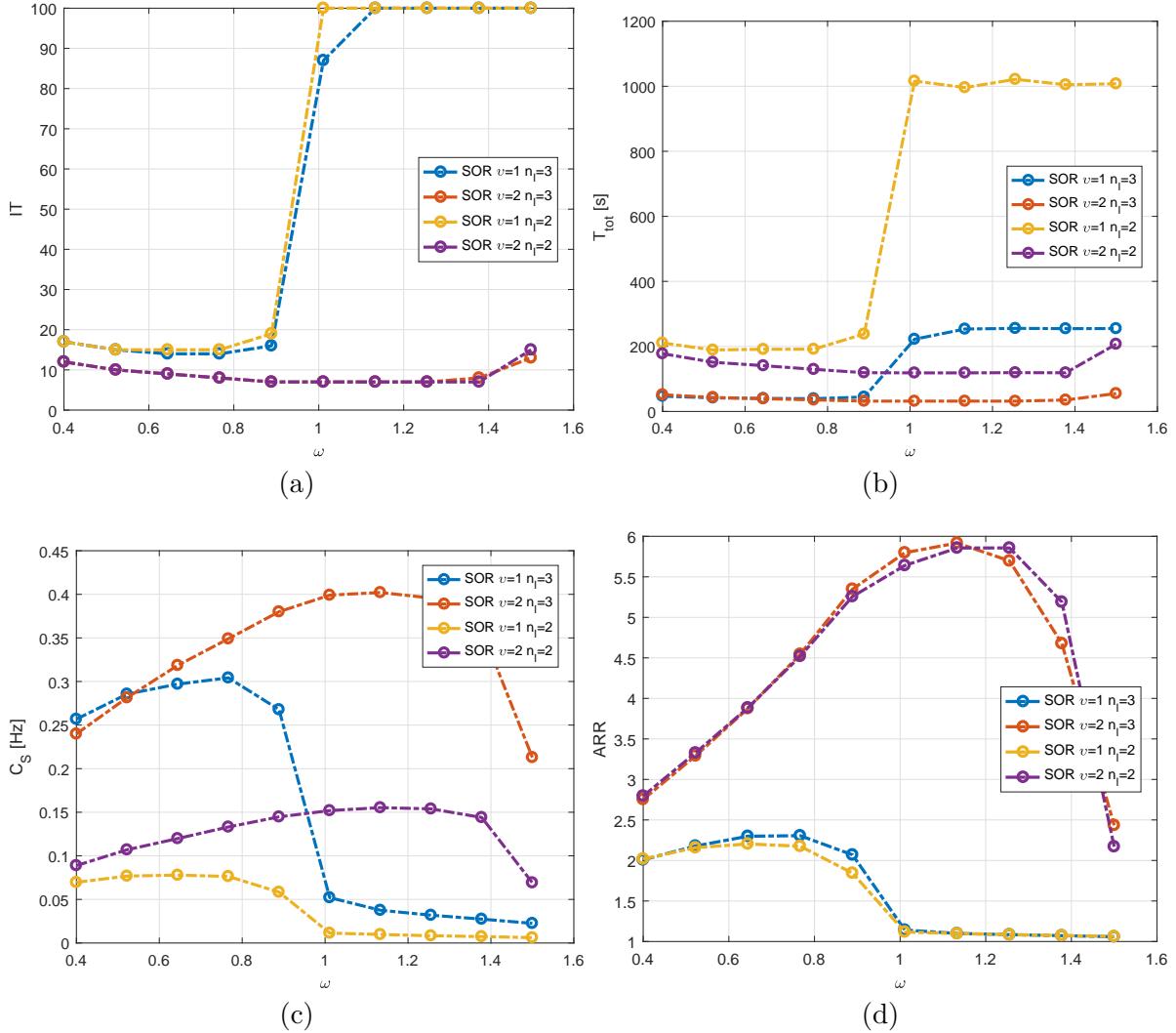


Figure 1.34: SOR smoother performances as a function of  $\omega$  and for different numbers of mesh levels  $n_l$ , different numbers of sweeping  $v$ . (a) Iterations at convergence.(b) CPU time [s]. (c) Convergence speed defined in equation [Hz] (1.168). (d) Average Residual Reduction per iteration defined in equation (1.167).

were reported in figure 1.34. In figures 1.34a and 1.34b one can observe a strong dependency of the convergence with the number of sweeping. In fact for  $v = 1$  and for  $\omega \geq 1$  PCG didn't converge within 100 iterations. The best convergence is obtained for  $\omega \approx 1.13$ ,  $v = 2$  and  $n_l = 3$  with 7 iterations in 31 seconds. In figures 1.34c and 1.34d one can also observe that for this point the convergence speed is of  $C_S \approx 0.4\text{Hz}$  and the  $ARR \approx 6$ . This investigation seems to suggest that one could improve performances increasing the number of levels and increasing the number of sweeping. Ichol smoother performances are described in figure 1.35. For this method the CPU time include the computation of incomplete Cholesky decompositions for each level. One can observe that much less PCG iterations are needed up to convergence

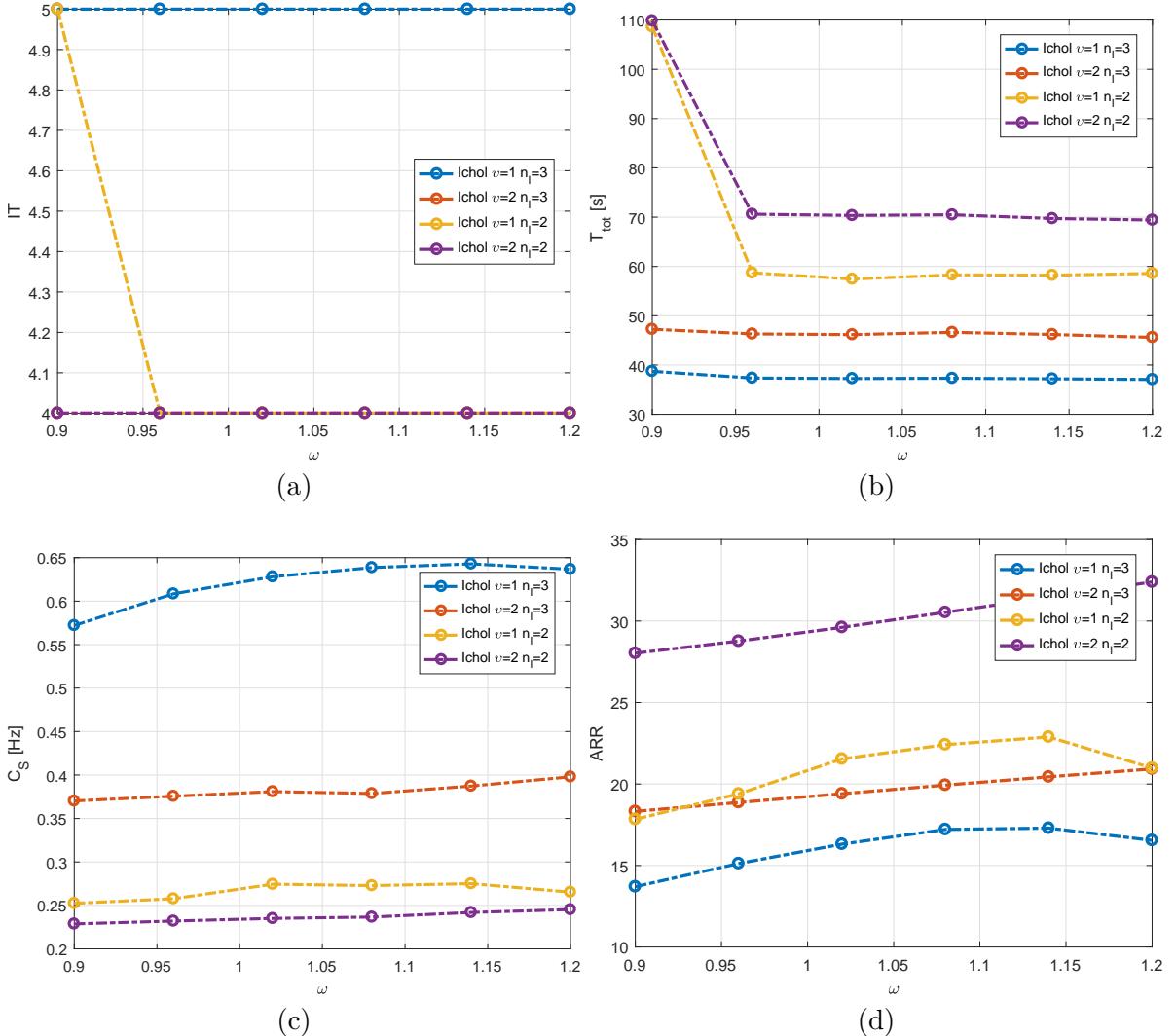


Figure 1.35: Ichol smoother performances as a function of  $\omega$  and for different numbers of mesh levels  $n_l$ , different numbers of sweeping  $v$ . (a) Iterations at convergence.(b) CPU time [s]. (c) Convergence speed defined in equation [Hz] (1.168). (d) Average Residual Reduction per iteration defined in equation (1.167).

( $\leq 5$ ). The CPU time per iteration is on the other hand very high and reaches its minimum of 37 seconds for  $\omega \approx 1.12$   $n_l = 3$  and  $v = 1$ . The maximum convergence speed is reached for the same point  $C_S \approx 0.64\text{Hz}$  and is bigger than the value found for the other methods. This means that Ichol smoother has a fast convergence, but still the investment required for the construction of preconditioning is not justified for the accuracy of  $10^{-5}$ . That is why finally the SOR optimal point is still better than the Ichol point if one looks at the total CPU time. If one considers the Line Search variants for Damped Jacobi and Ichol smoother (cf. table 1.1) It can be observed that the fastest combinations are for  $v = 1$  and  $n_l = 3$ . The best approach found for this configuration was the LS variant of Ichol with  $T_{tot} \approx 37$  seconds. Still this approach is less interesting then Ichol in terms of convergence speed. For a general conclusion

based on this configuration, depending on the accuracy needed ichol and LS-ichol smoothers can be effective for applications where strict accuracy is demanded so that the investment on the initial evaluation of the incomplete Cholesky decomposition is repaid by the improved convergence speed. On the other hand SOR and damped Jacobi are very inexpensive and can be adopted in situations where the final accuracy is less critical. Another important criterion of selection that is out of the scope of this work is the scalability of the preconditioner.

Table 1.1: Performance of Line Search variants

Method	$v$	$n_l$	$IT$	$T_{tot}[s]$	$C_S[Hz]$	$ARR$
LS-Damped Jacobi	1	2	12	53.1	0.2348	2.78
LS-Damped Jacobi	2	2	11	169.9	0.0892	2.89
LS-Damped Jacobi	1	3	18	50.5	0.2421	1.95
LS-Damped Jacobi	2	3	17	208.8	0.0699	2.01
LS-Ichol	1	2	4	52.9	0.3384	23.1
LS-Ichol	2	2	4	114	0.2348	39.4
LS-Ichol	1	3	4	37	0.5452	17.8
LS-Ichol	2	3	4	99.5	0.2635	23.5

## 1.5 Summary and conclusions

In this chapter the Finite Element Model approach was reviewed for the resolution of linear elastostatics problems. The optimization framework that is developed in this PhD requires in fact the reliable and fast evaluation of displacements, tip-clearance and stress in a large finite element model including both design space and engine models. In the introduction the challenges that this simulation problem represents were detailed. The engine model can be large and complex, to make it available in the proposed Matlab framework, the exploitation of superelements was first reviewed. The design space being meshed in an external environment, 8 node full integration hexahedral elements were reviewed. This gives the freedom to mesh mappable design space geometries that can easily satisfy aerodynamic and integration constraints. The communication between design space mesh and engine model is delicate since discretizations are not consistent at the interface. Several techniques were reviewed and compared with a proposed one. To make the evaluation of model responses faster in the optimization loop, multigrid preconditioned conjugate gradient was implemented and tested with several smoothers. To summarize the review in this chapter recovered the following topics :

- Finite Element notation
- Mesh tying
- Superelements
- Multigrid preconditioners

A personal contribution was given in:

- Weighted Average Continuity Approach (WACA) and Moment correction were proposed to deal with mesh tying related issues. Their performances were compared with reviewed approaches on simple 3D examples. Moment correction sensibly improves displacement or stress accuracy of all reviewed technique. Its implementation is recommended in Structural Finite Element models with mesh tying to enforce the local mechanical moment balance at the tied interface.
- This work led to the publication of a journal article [[Coniglio et al., 2018a](#)].
- Different smoother techniques were reviewed and benchmarked on a demonstration example that includes non-uniform hexahedral elements and the engine superelement for a 3 right hand side vectors application.

The studies conducted in this chapter show that the proposed framework, using a combination of reviewed techniques and proposed approaches, provides accurate and relatively fast responses that can be considered consistent with modeling hypothesis. As possible axes of improvement one can cite:

- Distributed memory implementations of the provided framework could also be considered as a valid improvement axis.
- Other important issue like shear locking control could be introduced in order to improve finite element accuracy.
- Still, the major limitation of the proposed framework consists in the fact that the WEM needs to be a linear model to be imported as a superelement. A possible solution in case of nonlinear engine models could be provided by non-intrusive coupling techniques [[Gendre et al., 2009](#)].

# Eulerian Structural Topology optimization

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## Résumé

Dans ce chapitre le design des structures conduites par simulation est passé en revue. Cela nécessite de 3 briques fondamentales:

- Le modèle d'analyse éléments finis.
- Le calcul des gradients.
- L'algorithme d'optimisation

Ces composantes sont représentées par des diagrammes par blocs, pour décrire leur interaction dans la boucle de design. La méthode adjointe de calcul des gradients et les algorithmes d'approximation séquentielle convexe sont aussi passés en revue. La méthode du Matériau Solide avec Pénalité (SIMP), représentant des approches Eulériennes d'optimisation topologique, est aussi décrite en détails. Pour cette méthode la solution est représentée par un champ de densité constant par élément de maillage. Les variables d'optimisation sont donc les pseudo-densités dans chaque élément fini dans la zone de design, qui peuvent prendre de valeurs entre 0 et 1. Ce qui correspond aux propriétés du vide et du solide respectivement. Pour éliminer les densités intermédiaires dans la solution, une pénalité est appliquée à la loi d'interpolation du module de Young entre le solide et le vide. Cette formulation souffre de plusieurs problèmes numériques. Cependant ceux-ci peuvent être traités par des techniques de restriction. Une approche multi-maillage pour un calcul efficace de la matrice de filtrage a été proposée pour réduire le cout d'évaluation de la matrice de filtrage. Ensuite pour traiter des contraintes de stress dans l'optimisation topologique l'approche unifiée d'agrégation relaxation [Verbart et al., 2017] est passé en revue et implémentée. Dans la dernière section de ce chapitre nous montrons l'application de l'approche SIMP au problème de design du mât et des attaches moteur. Des contraintes de symétrie de la solution et plusieurs cas de chargement sont également considérés.

Design activity is a complex human activity that relies on both quantitative and qualitative judgment. A product will be the consequence of several decisions that are made to respect legal, environmental, safety and economic requirements. Moreover generally speaking a product will be the "best" or at least satisfying with respect to its performances among some configurations. So the first requirement for the design process is being able to generate and describe design configurations or candidates. The second requirement is being able to evaluate the performance of a given candidate. Then the ranking of all configurations is made using one or several criteria. The concept of best design depends on the particular criterion chosen for ranking different configurations. Moreover these criteria depend on the decision maker's point of view. In fact performance depends on the particular level at which the design choice is made. For example in complex systems, the design choice that would benefit the most a particular component could be detrimental for the whole assembly's performance<sup>1</sup>. The person charged of making the choice about design, his position and his company could also have an impact on the criteria. For instance the recurrent costs and product lifetime span could be of primary importance for the consumer. On the other hand manufacturing costs

<sup>1</sup>For instance in aircraft many subcomponents need to be designed. Changing the design of one subcomponent will have an impact on the load path definition through other subcomponents. For this reason in the design loop finite element analysis are adopted to compute the load passing through each subcomponent interface.

could be more important for the manufacturer. Once the performance or cost criteria are defined, and all requirements are listed the choice would be to select the design that maximizes a given performance while respecting design constraints. The most naive approach to the design problem, would consist in producing each one of the design variants, test them and to selecting the best among them. This approach is of course impractical, expensive and incompatible with nowadays' economic reality. That's where modeling and simulation take their place. A model is any virtual or material abstraction of a given system that tries to predict its physical behavior with respect to particular operating conditions. This large definition can include any kind of simulation, or prototype that can be used to make a prediction on the configuration's behavior. It must be underlined that any model is only an abstraction of a particular phenomenon and not the real one. Every time a model is built some assumptions have to be considered, restricting of course it's domain of applicability<sup>2</sup>. Once a model is available the aforementioned approach could be used to select the best candidates among the proposed candidates. Still this approach is applicable only if an exhaustive list of candidates is available, the model is valid for each configuration and the time and computational resources needed to evaluate these configurations are consistent with the designer's means. To avoid such a brute force approach to design, optimization algorithms can be adopted to make automated choices that improve a given configuration. These algorithms can generate a sequence of improved designs, which are simulated until a convergence is attained. The discipline that deals with the interaction between the optimization strategy and structural simulation is called structural optimization. In this large discipline one can find several families of optimization approaches. These can be classified in base of design assumptions, design variables nature and model assumptions ([Bendsøe and Sigmund, 2003]) as:

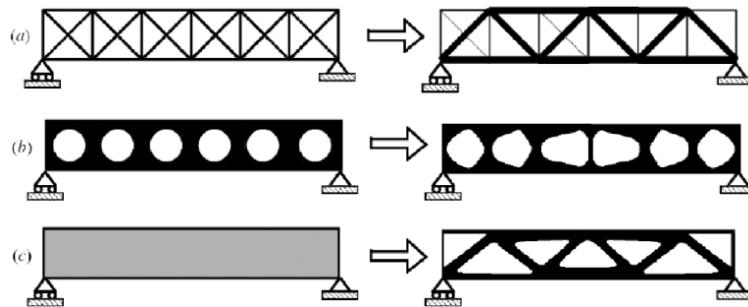


Figure 2.1: (a) Sizing,(b) shape and (c) topology optimization [Bendsøe and Sigmund, 2003]

- Size optimization: Often the design is mature, many modeling assumptions can be made. Thickness and cross section area typically are considered as design variables.
- Shape optimization: The design topology i.e. the number of holes in the structure is determined. Major modeling hypothesis can still be considered as true. The design variables are linked to the geometry in a freer manner. At this phase in fact both the contour of the domain and model can be modified by the design variables.

<sup>2</sup>With this term the author identifies the ensemble of all situations in which a model can be considered to behave closely to the real product

- Topology optimization: the design maturity is low, some If any assumptions can be made, these are only about the neighboring components. Often the boundary conditions' positions are considered fixed. Other consideration can enforce the material to be inside a volume that is often referred to as design zone. The design variables are linked with material density inside the design zone. Depending on the approach this link can be more or less straightforward.

In figure 2.1 sizing, shape and topology optimization are applied to the same problem. In size optimization the solution is assumed to be a truss. Bar cross sections are design variables, model connectivity is unchanged during optimization. In shape optimization, the solution is assumed to have 6 holes and to be described by a solid continuum model. Holes geometries are varied during optimization but holes cannot collapse as this would change model connectivity. In topology optimization the only hypothesis about the solution is that it can be represented by a solid continuum model. The number of holes and the connectivity of the solution are determined during the optimization. This chapter is mainly focused on topology optimization that is a very well established family of techniques in Structural Optimization, adopted for the determination of an optimal layout in preliminary design phases.

## 2.1 Simulation driven design

In this section some basic concepts, that will be used through the rest of this thesis, are introduced. This section is based on the first chapter of [Papalambros and Wilde, 2000].

### 2.1.1 Systems and Models representation

Any product or design can be represented as a system, that can be defined as an assembly of subsystems or components that perform a function or a process, which results in an output. Any system or process can then easily be represented by the block diagram of figure 2.2. For very complex products like aircrafts or gas turbine such a representation can help in

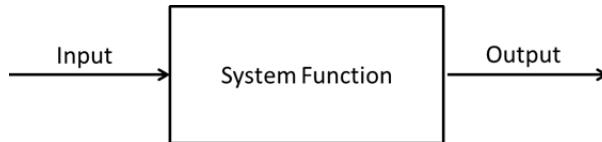


Figure 2.2: Block diagram representation

understanding the flow of inputs and outputs between each component. On the other hand system block diagrams are subjective and they depend on the particular phenomenon of interest. Moreover the system representation also depends on the level of complexity at which a product needs to be represented. To predict the behavior of such a system, models have to be built. A rigorous definition of a model according to [Papalambros and Wilde, 2000]:

*A model is an abstract description of the real world giving an approximate representation of more complex functions of physical systems.*

In the same way one can distinguish between *physical* models as the prototype that can emulate the real system's behavior with respect to some input and *symbolic* models that can emulate the system's behavior through simulation. In this thesis only *symbolic* models and more specifically *mathematical* models are considered:

*A mathematical model represents a system by mathematical relations [Papalambros and Wilde, 2000]*

Often this mathematical relation is explicitly available, or can be the result of a numerical computation. With respect to optimization, variables can be divided into:

- *Design Variables*: These quantities specify the design configuration. A design can be represented using such a variable to distinguish a design candidate from another.
- *Design Parameters*: These are quantities that identify the system's operating conditions. In structural models these can be identified with load and boundary conditions applied to the model.
- *Design constants*: These quantities are related to the physical phenomenon under study and cannot be altered. For example the gravity acceleration on the earth's surface.

This distinction is important because one can observe that the design of a system deals with the determination of *Design Variables*, under specific working conditions determined by *Design Parameters* and *Design constant* that are fixed for each design study.

### 2.1.2 Analysis model

As seen in chapter 1 structural FEMs are mathematical models that emulate the behavior of real structures under loading conditions. The input of such models will be the *Design Variables* that identify the structure studied. The outputs of these models are important responses that can have a more or less direct link with the engineering requirements. FEMs can in fact be used to determine safety factors, displacements, and others. For the linear static analysis model, one can consider a multi-input/multi-output block diagram representation that takes as input the *Design Variable* vector  $\{x\}$  and reads in output the displacement vector  $\{U\}$ , through the model's relation. Here the notation of Matrix of Right Hand Side (RHS) vectors  $[F(\{x\})]$  is employed for multiple loads, where each column corresponds to a load case. Moreover  $[U(\{x\})]$  is adopted for the corresponding solution vectors.

$$[K(\{x\})] [U(\{x\})] = [F(\{x\})] \quad (2.1)$$

It should be noticed that both stiffness and load vectors can be a function of the configuration i.e. of the *Design Variable*  $\{x\}$ . Moreover a FEA block hides several operations:

- The stiffness matrix assembly.

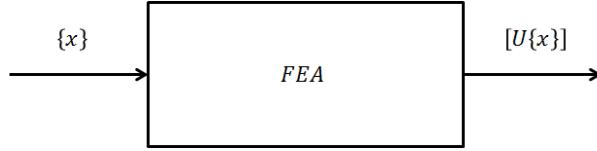


Figure 2.3: FEA block diagram representation

- The right hand side vectors assembly.
- The linear system solution per each right hand side vector.

This functional analysis allows a more detailed description of the FEA subsystem's block description c.f. figure 2.4:

After displacements are computed, a post-processing is often needed to evaluate responses

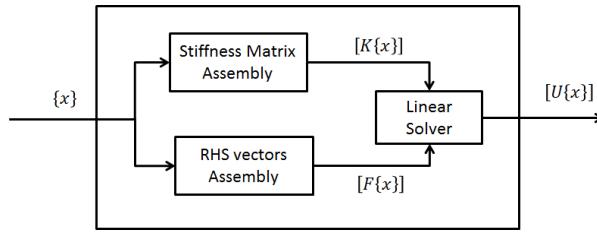


Figure 2.4: FEA subsystem block diagram representation

of interest that are needed for design selection. The post processing block representation is shown in figure 2.5. This system takes as input design variable's vector and the displacement vectors, and as output:

- Optimization cost functions  $\{o(\{x\})\}$ . In multi-criteria/multi-objective optimization problems it is in fact possible to have several functions that one want to minimize or maximize in the final design. Often it is possible to consider a singular objective function as several techniques exist to transform a multi-objective problem into a single-objective one [Marler and Arora, 2004].
- The inequality constraint vector  $\{g(\{x\})\}$ . This vector contributes to the problem's definition through inequalities that have to be respected by the design candidate in order to be accepted.
- The equality constraint vector  $\{h(\{x\})\}$ . This vector contributes to the problem's definition through equalities that have to be respected by the design candidate in order to be accepted.

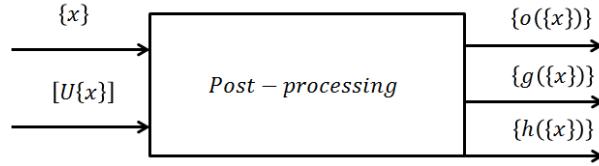


Figure 2.5: Post-processing block diagram representation

### 2.1.3 Optimization problem formulation

Generally an optimization problem will be formulated as follows:

$$\begin{cases} \min \{o(\{x\})\} \\ s.t. \\ \{x\} \in D \subset \mathbb{R}^n \\ g_i(\{x\}) \leq 0 \quad \forall i = 1, 2, \dots, m \\ h_j(\{x\}) = 0 \quad \forall j = 1, 2, \dots, l \end{cases} \quad (2.2)$$

where  $D$  is a subset of  $\mathbb{R}^n$ ,  $n$  is the number of design variables,  $m$  and  $l$  are respectively the number of inequality and equality constraints. Depending on the nature of  $D$  one can make a first distinction between continuous and integer programming optimization problems. In the first case each design variable belongs to a closed interval in  $\mathbb{R}$  which means that  $D \equiv \{\{x\} | \{l_b\} \leq \{x\} \leq \{u_b\}\}$ , where  $\{l_b\}$  and  $\{u_b\}$  are respectively design variable lower and upper bound vectors. For the second family of problems, the design variable belongs to a list of values.<sup>3</sup> Most real life design problems include a combination of continuous and integer design variables. This thesis mainly focused on continuous optimization problems. A second distinction between optimization problems can be introduced for the number of criteria considered as simultaneous goals. One can consider either single or multi-objective design problems. As aforementioned the case where a list of goals needs to be considered can be carried out by the solution of a group of single objective problems. Another hypothesis that will be considered for responses is that they will be regular at least up to the first derivative with respect to the design variables. This hypothesis allows the use of efficient gradient based techniques for the optimization problem solution.

### 2.1.4 Optimum characterization

A first important question about an optimization problem is about the existence of an optimal solution. In fact the feasible space, i.e. the space of solutions that simultaneously satisfy all constraints in the problem, can degenerate to a void space. Let's consider a 2D example

<sup>3</sup>Depending on the fact that one can sort or not design variable, one can also have the distinction between categorical or integer design variable

where the design vector  $\{x\} \equiv \{x_1, x_2\}^T$  has to solve the following optimization problem:

$$\begin{cases} \min(o(x_1, x_2)) = x_1 + x_2 \\ g_1(x_1, x_2) = \frac{x_1^2}{a^2} + \frac{x_2^2}{b^2} - 1 \leq 0 \\ g_2(x_1, x_2) = c - (c + x_1)x_2 \leq 0 \\ 0 \leq x_1 \leq 1 \\ 0 \leq x_2 \leq 1 \end{cases} \quad (2.3)$$

where  $a, b, c$  are parameters. For some choice of these parameters the feasible space can be void as it is shown in figure 2.6b. In such cases the solution to the optimization problem will

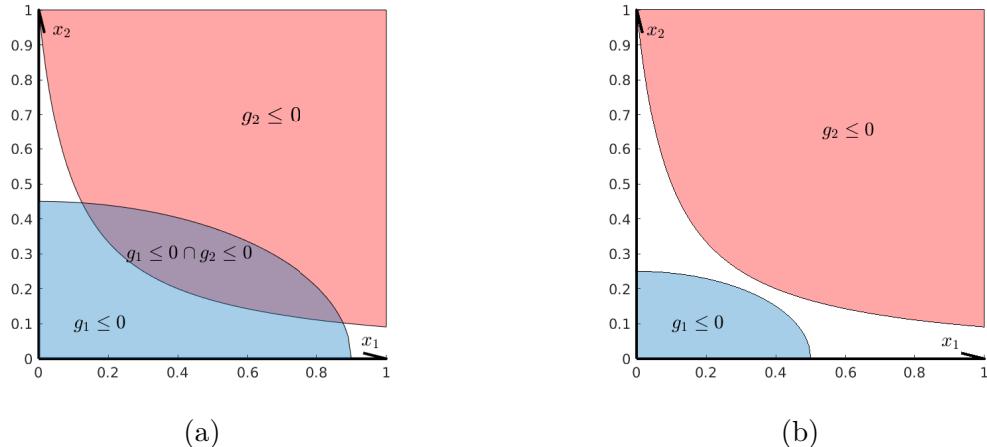


Figure 2.6: (a) Example of design domain representation for  $a = 0.9, b = 0.45$  and  $c = 0.1$ ; (b) degenerate feasible domain for  $a = 0.5, b = 0.25$  and  $c = 0.1$ ;

not exist. This situation often arises when dealing with antagonistic responses for constraints (for instance mass and maximum stress in a design). When this happens the only way of determining if a combination of requirements gives a void feasible design is by trials.

Another important property of an optimization problem is convexity. A convex optimization problem is one whose objective and constraints are convex. It can be shown that the strictly convex optimization problem has only one solution and this solution is the global optimum of the optimization problem. For a non-convex problem it is necessary to introduce the concept of local and global optimality. In fact a local optimum is a design configuration that is the best design for the goal function and that respects constraints, inside a small enough neighborhood of the solution. On the other hand a global optimum is the best design that satisfies the constraints over the whole design space and hence is the solution of the optimization problem. It is possible to show this issue with the following 1D example. Given the optimization problem:

$$\begin{cases} \min o(x) = \frac{3}{2}x^5 - \frac{5}{2}x^3 \\ s.t. \\ -\frac{3}{2} \leq x \leq \frac{3}{2} \end{cases} \quad (2.4)$$

The objective function is shown in figure 2.7. It is possible to observe that the point  $x = 1$  is

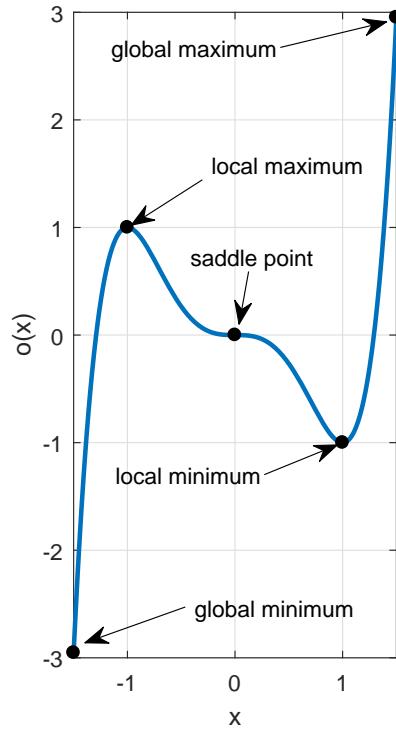


Figure 2.7: Optima characterization for the simple polynomial  $o(x) = \frac{3}{2}x^5 - \frac{5}{2}x^3$

a local minimum because it is possible to define a small interval around this point in which  $o(x)$  is minimum. On the other hand  $x = -\frac{3}{2}$  is the global minimum. Typically a non-convex problem (such as the one of equation (2.4)) will possess several local optima but only one global optimum. In general, finding the global optimum can be considered as an insurmountable challenge for most optimization problems. For this reason, often one can only conclude that a design is an improvement of an existing one, but is not the best design ever achievable. With respect to constraints, an optimum can be either on the inside (interior optimum) or on the boundary (boundary optimum) of the design domain. The first situation arises when constraints are inactivated. This also implies that the unconstrained optimization problem (obtained ignoring all constraints) has the same solution as the constrained optimization problem. When this is not the case, the optimum will lie on the feasible design boundary. This implies that at least one constraint will be active so that  $g_i = 0$  for at least one  $i \in \{1, 2, \dots, m\}$ . Here necessary and sufficient conditions to determine if a point is a local minimum in a

constrained problem are reviewed<sup>4</sup>. Given an optimization problem in the form of <sup>5</sup>

$$\begin{cases} \min o(\{x\}) \\ s.t. \\ \{h(\{x\})\} = \{0\} \\ \{g(\{x\})\} \leq \{0\} \end{cases} \quad (2.5)$$

Given the Lagrangian of the problem:

$$\mathcal{L}(\{x\}) := o(\{x\}) + \{\lambda\}^T h(\{x\}) + \{\mu\}^T g(\{x\}) \quad (2.6)$$

where  $\{\lambda\} \in \mathbb{R}^l$  and  $\{\mu\} \in \mathbb{R}^m$  are Lagrangian multipliers. The necessary optimality conditions also known as *Karush-Khun-Tucker (KKT) conditions* are:

$$\begin{cases} \{h(\{x\})\} = \{0\} \\ \{g(\{x\})\} \leq \{0\} \\ \nabla o + \{\lambda\}^T \nabla h + \{\mu\}^T \nabla g = \{0\} \\ \{\mu\} \geq \{0\} \\ \{\mu\}^T \{g(\{x\})\} = 0 \end{cases} \quad (2.7)$$

when a point satisfies these conditions is said to be a KKT point and may not be a minimum since the conditions are not sufficient. Second order information is necessary to verify the nature of a KKT point:

*If a KKT point  $\{x_*\}$  exists, such that the Hessian of the Lagrangian on the subspace tangent to the active constraints (equalities and inequalities) is positive-definite at  $\{x_*\}$ , then  $\{x_*\}$  is a local constrained minimum.*

Coming back to the proposed 1D example (2.4) it is possible to show that  $x_1 = -\frac{3}{2}, x_2 = -1, x_3 = 0$  and  $x_4 = 1$  are KKT points. In fact being  $\frac{do}{dx} = 0$  for  $x_2 = -1, x_3 = 0$  and  $x_4 = 1$  in all these points Lagrange multipliers associated with the bound constraints  $x + \frac{3}{2} \leq 0$  and  $-x + \frac{3}{2} \leq 0$ , respectively  $\mu_1$  and  $\mu_2$  are both equal to 0. Making the analysis of the Hessian of the Lagrangian (in this case  $\frac{d^2o}{dx^2} = 30x^3 - 15x$  being all constraints inactivated) one can observe that:

- $\frac{d^2o}{dx^2}(-1) = -15$ , so that  $x = -1$  is a local maximum
- $\frac{d^2o}{dx^2}(1) = 15$ , so that  $x = 1$  is a local minimum
- $\frac{d^2o}{dx^2}(0) = 0$ , so that nothing can be stated on the point in  $x = 0$ . Making an analysis of the third order derivative one can actually conclude that this KKT point is actually a saddle point.

For the point in  $x = -\frac{3}{2}$  the second constraint is inactive so that  $\mu_2 = 0$ . On the other hand being  $g_1 = 0$ ,  $\mu_1$  can be different than zero (or not). Using the KKT conditions one

---

<sup>4</sup>For the proof the interested reader can find it in [Papalambros and Wilde, 2000].

<sup>5</sup>Bound constraints on design variables can be considered as part of  $\{g(\{x\})\} \leq \{0\}$

can obtain  $\mu_1 = \frac{d\phi}{dx}(-\frac{3}{2}) = \frac{675}{32} > 0$ . Being the number of variables equal to the number of active constraints the plane tangent to the active constraint is degenerated to  $\delta x = 0$ . In this case it is simpler to consider the fact that  $\phi(x)$  is strictly monotone in the neighbor of  $x = -\frac{3}{2}$ . Moving from this point in the feasible space, the objective function will be increased. This means that  $x_1 = -\frac{3}{2}$  is a local minimum. Moreover being  $\phi(x_1) = -\frac{189}{64} < \phi(x_4) = -1$  one can conclude that  $x_1$  is a global minimum. This kind of analysis makes the optimization problem solvable because both objective and constraint functions are analytic and inexpensive. Responses computed from the results of simulations can take much more time. That is why optimization algorithms need to be considered.

### 2.1.5 Optimization solvers

An optimization solver is a numerical strategy that can be adopted to generate a succession of designs that converge to at least a local optimum. Several approaches exist in the literature and can be chosen to treat a given numerical problem. Some important families of approaches can be determined depending on:

- The information about the derivatives of design responses needed by the optimizer to provide an improved design. One can say that an optimizer is of order  $p$  if it uses up to the  $p^{th}$  derivatives information to generate a new optimal candidates. A solver that uses only the function values is a 0 order method. On the other hand methods that use both the information of the function and the one of the gradients are considered first order methods and so on.
- The priority given to design exploration with respect to the local optima refinement. A local optimization solver, aims at improving an initial design without any claiming about the global optimality. On the other hand global optimization solvers also explore the design space and often are proven to converge to the global optimum if the optimization iteration goes to infinity.
- The fact of adopting or not random processes inside the optimization loop. A deterministic solver starting from the same initial point will always fall in the same local minimum, when repeating the optimization process. In semi-stochastic solvers like evolutionary algorithms [Simon, 2013] initializing the algorithm with the same starting individual (or population) the results can be different when repeating the optimization process.
- The fact of using the information coming from one individual or several individuals simultaneously. Surrogate assisted optimizations [Forrester et al., 2008] are an example of approaches that use multi-design information. In these approaches approximations of the real objective and constraint functions are built on the basis of an initial design of experiments. Then the surrogates can be used to find new optimal candidates. Genetic Algorithms and other evolutionary algorithms also combine information coming from a population of individuals.

This thesis first order optimization solvers were employed. In particular Sequential Convex Programming algorithms such as the Method of Moving Asymptotes were implemented [Svanberg, 1987, Svanberg, 2002]. From a block diagram point of view such optimizer can be seen as a system that takes as input the design variable vector, goal and constraint responses and their sensitivity at a given point and reads in output an improved design configuration (cf. figure 2.8). In this representation only inequality constraints were considered inside the

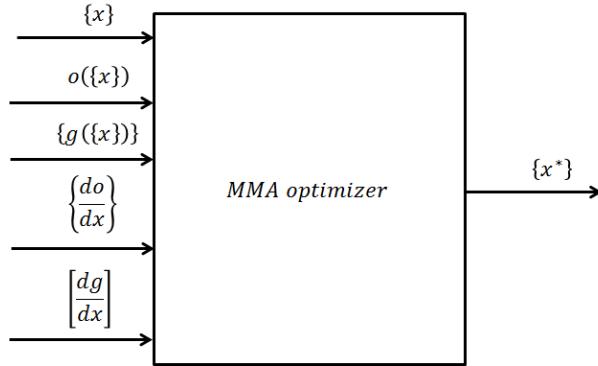


Figure 2.8: MMA optimizer block diagram representation

optimization problem.

### 2.1.6 Sequential Convex Programming algorithms

In this subsection Sequential Convex Programming (SCP) [Fleury, 1993, Zillober et al., 2004, Duysinx et al., 2009] general idea is described in more details. In this family of approaches it is possible to cite convex linearization (CONLIN) algorithm [Fleury and Braibant, 1986], the Method of Moving Asymptotes (MMA) [Svanberg, 1987, Svanberg, 2002], and their variants like [Borrvall and Petersson, 2001, Bruyneel et al., 2002] and [Zillober et al., 2004]. Recently [Etman et al., 2012] provided a review of SCP methods and discussed the fact that approximated approximation approach [Groenwold et al., 2010] can be interpreted as diagonal quadratic subproblem in the spirit of the well-known SQP algorithms [Nocedal and Wright, 2006]. SCP algorithms are all based on reciprocal or reciprocal-like approximations of the original non-linear programming problem (c.f. equation (2.2)). They generate in this way a series of convex non-linear programming subproblems. To generate such subproblems the following steps are followed:

1. Reciprocal intervening variables are replaced into first order Taylor series expansions of both objective and constraint responses
2. The problem is made more convex than as aforementioned ensure the uniqueness of subproblems solution.
3. Often convexity is adjusted on the base of convergence history.

When these approximations are built at the  $k^{th}$  iteration of the optimization loop, the  $P[k]$  subproblem has the following form.

$$\begin{cases} \min_{\{x\}} \tilde{o}^{[k]} \\ s.t. \\ \{\tilde{g}\}^{[k]} \leq 0 \\ \{l_b\} \leq \{x\} \leq \{u_b\} \end{cases} \quad (2.8)$$

The approximations adopted by MMA and CONLIN are separable (i.e. Hessian is diagonal). This means that efficient subproblem solvers can be adopted [Falk, 1967, Fleury, 1979]. In order to ensure the convergence of  $P[k]$  subproblem solutions  $\{x\}_*^{[k]}$  to a solution (at least local) of the original problem (c.f. equation 2.2), either trust region [Alexandrov et al., 1998] strategies either conservatism frameworks [Svanberg, 2002] can be adopted [Etman et al., 2012]. In the first case the convergence is guaranteed introducing an allowed search domain around the iteration point  $\{x\}^{[k]}$ , i.e. the following constraint:

$$\{x\} \in \mathcal{C}^{[k]} \equiv \{-\delta^{[k]} \leq \{x\} - \{x\}^{[k]} \leq \delta^{[k]}\} \quad (2.9)$$

is added to the problem's formulation. In the conservatism framework, the convexity of both cost and constraint approximations is adjusted to enforced convergence. Moreover the solution of each  $P[k]$  subproblem  $\{x\}_*^{[k]}$  in both cases may be rejected. In such case the subproblem is built again this time with a smaller trust region or with more conservative approximation until either convergence is reached or until the point  $\{x\}_*^{[k]}$  can be accepted as new iteration point  $\{x\}^{[k+1]}$  [Svanberg, 2002].

### 2.1.6.1 The Method of Moving Asymptotes

Here MMA derivation from intervening variable introduction and the implementation details of [Svanberg, 2007] are detailed. In this approach reciprocal intervening variables are introduced in the first order Taylor expansion. For several structural optimization problems it is known that this may yield nonlinear approximations of significantly better accuracy than linear [Etman et al., 2012]. Therefore let's consider the first order Taylor expansion of the objective function with respect to a generic intervening variables  $y_i(x_i)$ :

$$\tilde{o}(\{y\}) = o(\{y\}^{[k]}) + \left\{ \frac{\partial o^{[k]}}{\partial y} \right\}^T (\{y\} - \{y\}^{[k]}) \quad (2.10)$$

MMA makes the use of the following intervening variables:

$$y_i = \begin{cases} \frac{1}{x_i - l_i^{[k]}} & \text{if } \frac{\partial o^{[k]}}{\partial x_i} < 0 \\ \frac{1}{u_i^{[k]} - x_i} & \text{if } \frac{\partial o^{[k]}}{\partial x_i} > 0 \end{cases} \quad (2.11)$$

where  $l_i^{[k]}$  and  $u_j^{[k]}$  are the lower and upper asymptotes. Equation (2.11) substituted in (2.10) gives:

$$\tilde{o}(\{x\}) = \sum_{j=1}^N \left( \frac{p_j^{[k]}}{u_j^{[k]} - x_j} + \frac{q_j^{[k]}}{x_j - l_j^{[k]}} \right) + r^{[k]} \quad (2.12)$$

where

$$p_j^{[k]} = (u_j^{[k]} - x_j^{[k]})^2 \left( \frac{\partial o^{[k]}}{\partial x_j} \right)^+ \quad (2.13)$$

$$q_j^{[k]} = (x_j^{[k]} - l_j^{[k]})^2 \left( \frac{\partial o^{[k]}}{\partial x_j} \right)^- \quad (2.14)$$

$$r^{[k]} = o(\{x\}^{[k]}) - \sum_{j=1}^N \left( \frac{p_j^{[k]}}{u_j^{[k]} - x_j^{[k]}} + \frac{q_j^{[k]}}{x_j^{[k]} - l_j^{[k]}} \right) \quad (2.15)$$

$$\left( \frac{\partial o^{[k]}}{\partial x_j} \right)^+ = \max \left( 0, \frac{\partial o^{[k]}}{\partial x_j} \right) \quad (2.16)$$

$$\left( \frac{\partial o^{[k]}}{\partial x_j} \right)^- = \max \left( 0, -\frac{\partial o^{[k]}}{\partial x_j} \right) \quad (2.17)$$

It is useful to consider their visualization for a 1D example (c.f. figure 2.9) In such case it is

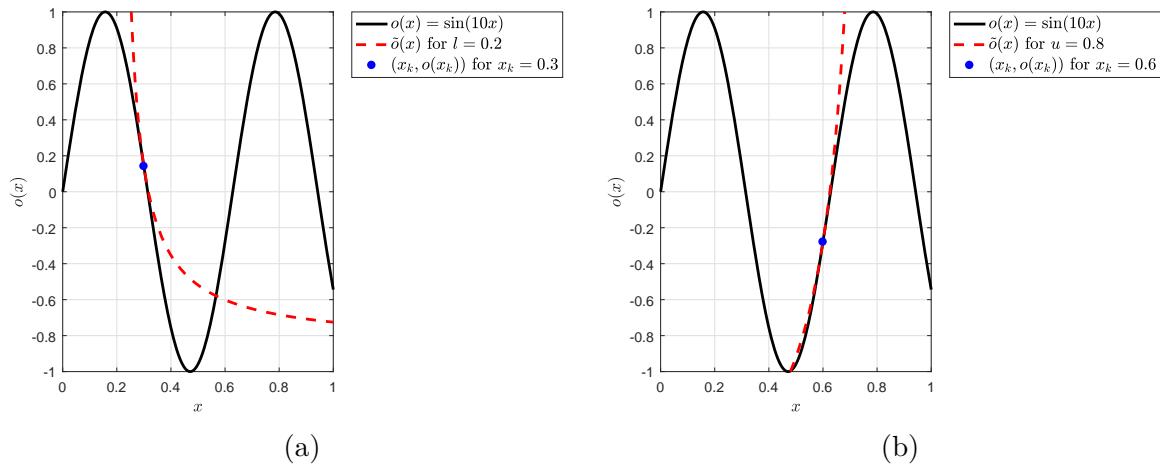


Figure 2.9: MMA approximations functions for a simple 1D example. The functions have different form depending on the derivative sign at the iteration point. The more asymptotes are close to the iteration point the more convex is the approximation function.

possible to recognize that the approximation function being first order approximation of the original function, shares with it its value and its derivative at the iteration point. It is also important to link the effect of asymptotes position on the approximation function and the impact that this can have on MMA conservatism. Let's consider again the elliptic feasible

space described by:

$$g(x, y) = \frac{x^2}{a^2} + \frac{y^2}{b^2} - 1 \leq 0 \quad (2.18)$$

the asymptotes in each direction can here be put at a constant infinity norm distance from the current point:

$$u_x - x_k = u_y - y_k = x_k - l_x = y_k - l_y = \Delta_a \quad (2.19)$$

The effect of the variation of  $\Delta_a$  on the MMA approximation  $\tilde{g}(x, y)$  is shown in figure 2.10. It is possible to observe that smaller values of  $\Delta_a$  reduce the size of the approximated feasible

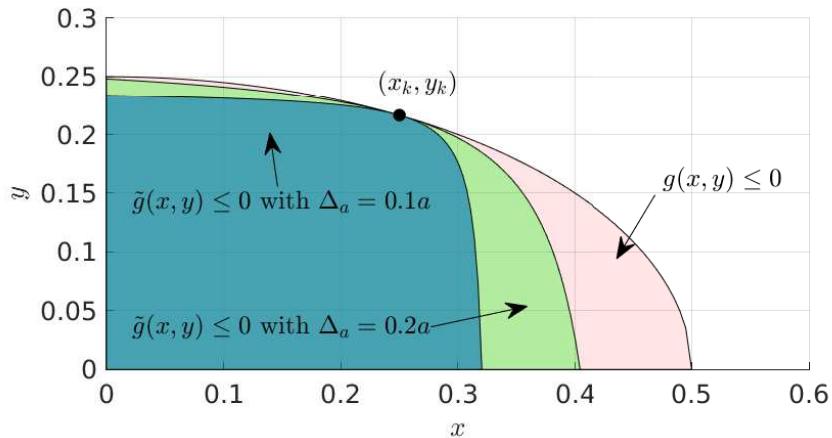


Figure 2.10: Effect of the asymptotes distance  $\Delta_a$  from the current point  $(x_k, y_k) \equiv \left(\frac{a}{2}, \frac{3}{2}b\right)$  on MMA approximation function  $\tilde{g}(x, y)$  of the original constraint function  $g(x, y)$  ( $a = \frac{1}{2}, b = \frac{1}{4}$ ). The smaller the distance, the higher is the approximation convexity, the smaller is the approximated feasible design space, the more conservative MMA will behave.

design space (i.e.  $\tilde{g}(x, y) \leq 0$ ). In such way MMA solver can produce conservative subproblem optima that are more prone to satisfy true constraints (i.e.  $g(x, y) \leq 0$ ). Understanding this principle is essential to correctly tune the solver parameters as it is described in appendix C.3. It can be observed that the presented version of MMA actually fails encountering cost function or objective saddle points. For this reason in the version of [Svanberg, 2007] equations

(2.13)-(2.14) are modified as follows:

$$p_j^{[k]} = (u_j^{[k]} - x_j^{[k]})^2 \left( 1.001 \left( \frac{\partial o^{[k]}}{\partial x_j} \right)^+ + 0.001 \left( \frac{\partial o^{[k]}}{\partial x_j} \right)^- + \frac{\rho_j}{u_{bj} - l_{bj}} \right) \quad (2.20)$$

$$q_j^{[k]} = (u_j^{[k]} - x_j^{[k]})^2 \left( 0.001 \left( \frac{\partial o^{[k]}}{\partial x_j} \right)^+ + 1.001 \left( \frac{\partial o^{[k]}}{\partial x_j} \right)^- + \frac{\rho_j}{u_{bj} - l_{bj}} \right) \quad (2.21)$$

where  $\rho_j$  is a parameter, by default set to  $10^{-5}$  that enforce strictly convexity of approximations [Svanberg, 2007]. The asymptotes position is adjusted during optimization to make MMA behavior more or less conservative. As default for the first two iterations, asymptotes positions are initialized as:

$$l_j^{[k]} = x_j^{[k]} - 0.5(u_{bj} - l_{bj}) \quad (2.22)$$

$$u_j^{[k]} = x_j^{[k]} + 0.5(u_{bj} - l_{bj}) \quad (2.23)$$

in later iterations they evolve as:

$$l_j^{[k]} = x_j^{[k]} - \gamma_j^{[k]} (x_j^{[k-1]} - l_j^{[k-1]}) \quad (2.24)$$

$$u_j^{[k]} = x_j^{[k]} + \gamma_j^{[k]} (x_j^{[k-1]} - l_j^{[k-1]}) \quad (2.25)$$

where

$$\gamma_j^{[k]} = \begin{cases} 0.7 & \text{if } (x_j^{[k]} - x_j^{[k-1]})(x_j^{[k-1]} - x_j^{[k-2]}) < 0 \\ 1 & \text{if } (x_j^{[k]} - x_j^{[k-1]})(x_j^{[k-1]} - x_j^{[k-2]}) = 0 \\ 1.2 & \text{if } (x_j^{[k]} - x_j^{[k-1]})(x_j^{[k-1]} - x_j^{[k-2]}) > 0 \end{cases} \quad (2.26)$$

provided that this leads to values that satisfy:

$$l_j^{[k]} \leq x_j^{[k]} - 0.01(u_{bj} - l_{bj}) \quad (2.27)$$

$$l_j^{[k]} \geq x_j^{[k]} - 10(u_{bj} - l_{bj}) \quad (2.28)$$

$$u_j^{[k]} \geq x_j^{[k]} + 0.01(u_{bj} - l_{bj}) \quad (2.29)$$

$$u_j^{[k]} \leq x_j^{[k]} + 10(u_{bj} - l_{bj}) \quad (2.30)$$

If any of this value is violated the corresponding  $l_j^{[k]}$  or  $u_j^{[k]}$  is put to the right hand side of the violated inequality [Svanberg, 2007]. The asymptotes update strategy proposed in this approach in some cases become too aggressive with respect to constraint violations and sometimes diverge. For this reason globally convergent version of MMA was proposed in [Svanberg, 2002]. Despite the risk of divergence, the simple "always accept" strategy is nevertheless frequently adopted and found to be effective and efficient in many structural optimization applications [Groenwold and Etman, 2010]. That is why such version was also considered in the present thesis even though specific tuning is proposed to enforce MMA conservatism (c.f. appendix C.3).

### 2.1.7 Gradient adjoint evaluation

The MMA optimizer takes as input all responses values and derivatives with respect to design variables, lower and upper bounds and gives as output the vector of the new optimal candidate. The success of MMA algorithm is therefore strictly linked to the accuracy of derivatives provided. In order to efficiently compute response gradients many approaches can be found in the literature [Van Keulen et al., 2005] [Martins, 2012]. For explicit functions symbolic differentiations is possible by hand or using appropriate software like Maple, Mathematica or Matlab. Nevertheless for general algorithms it may be impossible to produce a closed form expression for the derivatives. Finite differences are always an easily available option when a black box model is provided even though their efficiency is far from satisfactory. In fact several evaluations of responses are required and truncation error can also have an impact on the evaluation accuracy. Complex-step derivative approximation [Lyness and Moler, 1967] are also efficient strategy that can be adopted when one have access to the core code of the provided response. The only requirement for this approach is that the model should be able to process complex input. As for finite difference, the number of design variable has a significant effect on the computational burden of this approach and therefore is impractical in the case of topology optimization problems. Algorithmic Differentiation (AD) [Griewank, 2000, Naumann, 2012] consists in the systematic application of differentiation chain rule to computer programs. It is as accurate as analytic method and can be much easier to implement [Martins, 2012]. Still the most accurate and efficient methods available are analytic methods [Martins, 2012]. In this family of approaches one can find both direct and adjoint approaches. The first is more advantageous for optimization problem where the number of design variables is smaller than the number of responses which sensitivities need to be computed, otherwise the adjoint is the most efficient strategy as the number of right hand side vector to be considered in the sensitivity evaluation is reduced. Being topology optimization characterized by a much greater number of design variables compared to the number of responses, here the adjoint evaluation is considered. To this purpose let's consider a generic response  $O(\{x\}, [U(\{x\})])$  that depends both directly and through the displacement vector on the design variables.

$$\left\{ \frac{dO}{dx} \right\} = \left\{ \frac{\partial O}{\partial x} \right\} + \sum_{j=1}^{N_l} \left[ \frac{dU_j}{dx} \right] \left\{ \frac{\partial O}{\partial U_j} \right\} \quad (2.31)$$

By taking the derivatives of equations (2.1) (for the  $j^{th}$ -load case) one gets:

$$\frac{d[K(\{x\})]}{dx_i} \{U_j(\{x\})\} + [K(\{x\})] \frac{d\{U_j(\{x\})\}}{dx_i} = \frac{d\{F_j(\{x\})\}}{dx_i} \quad (2.32)$$

Writing the product  $\frac{d[K(\{x\})]}{dx_i} \{U_j(\{x\})\}$  as columns of the matrix  $\left[ \frac{dK}{dx} U_j \right]$  and  $\frac{d\{F_j(\{x\})\}}{dx_i}$  as the columns of the matrix  $\left[ \frac{dF_j}{dx} \right]$  one can also write:<sup>6</sup>

$$\left[ \frac{dU_j}{dx} \right]^T = [K]^{-1} \left( - \left[ \frac{dK}{dx} U_j \right] + \left[ \frac{dF_j}{dx} \right] \right) \quad (2.33)$$

---

<sup>6</sup>Hereafter the dependency on  $\{x\}$  and  $\{U_j(\{x\})\}$  is neglected for conciseness

Defining the adjoint vector as:

$$\{\beta_j\} = [K]^{-1} \left\{ \frac{\partial O}{\partial U_j} \right\} \quad (2.34)$$

One can finally rewrite equation (2.31):

$$\left\{ \frac{dO}{dx} \right\} = \left\{ \frac{\partial O}{\partial x} \right\} + \sum_{j=1}^{N_l} \left( - \left[ \frac{dK}{dx} U_j \right]^T + \left[ \frac{dF_j}{dx} \right]^T \right) \{\beta_j\} \quad (2.35)$$

Such evaluation requires the a solution of static balance equations once per each load case and per each response. This still represents an improvement for topology optimization problems where the number of design variables is much larger than the number responses times the number of load cases. It is possible to think about the adjoint evaluation of sensitivity as

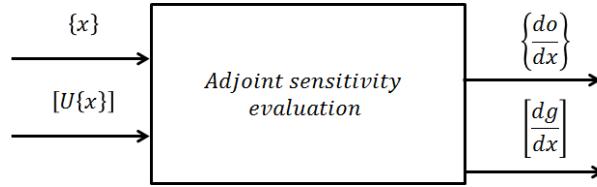


Figure 2.11: Adjoint sensitivity evaluation block diagram representation

a model itself. This model takes as input the design variable's vector and the displacement vectors and reads in output the response sensitivity (cf. figure 2.11). It must be noticed that some special responses, may not require any supplementary solution of static balance equations for the sensitivity evaluation. This is the case of compliance response. The compliance or the strain energy is commonly adopted as optimization objective in topology optimization. It is defined with respect to the  $j$ -th load case as:

$$c_j(\{x\}) = \{U_j(\{x\})\}^T [K(\{x\})] \{U_j(\{x\})\} = \{U_j(\{x\})\}^T \{F_j(\{x\})\} \quad (2.36)$$

For such response one can easily compute:

$$\left\{ \frac{\partial c_j}{\partial x} \right\} = \left[ \frac{\partial F_j}{\partial x} \right]^T \{U_j(\{x\})\} - \left[ \frac{\partial K}{\partial x} U_j \right]^T \{U_j(\{x\})\} \quad (2.37)$$

$$\left\{ \frac{\partial c_j}{\partial U_j} \right\} = \{F_j(\{x\})\} \quad (2.38)$$

$$\{\beta_j\} = \{U_j(\{x\})\} \quad (2.39)$$

As aforementioned the adjoint vector being equal to the displacement vector, does not require any additional system solution for its evaluation. This results in a large computational economy for such design responses.

### 2.1.8 Design loop & human interaction

In the previous subsections each block inputs and outputs were defined, in this section is going to show how these are connected into the design loop, and how the automated design loop is related with the human intervention. In figure 2.12 a complete structural optimization

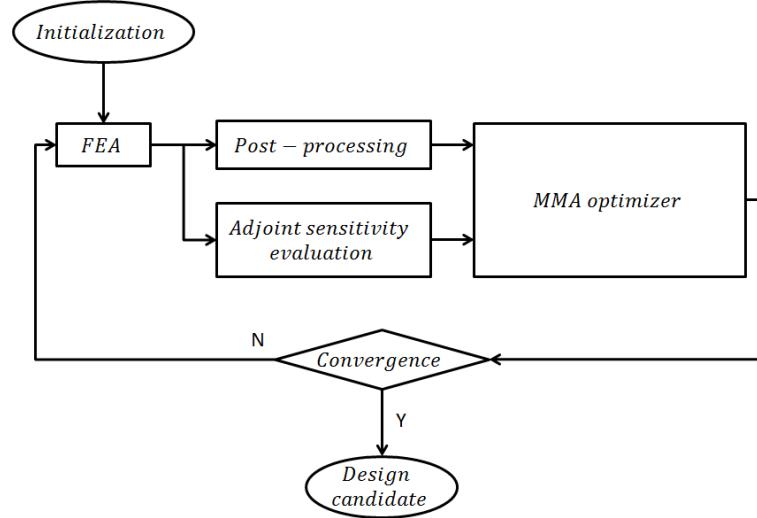


Figure 2.12: Optimization loop block diagram representation.

loop is illustrated. In the first phase the design loop has to be initialized with a starting guess design. This guess can be randomly selected. Anyway it is a good practice, when it is possible, to provide a starting guess that respects all constraints. This avoids useless optimization iterations required to reach the feasible domain. Another reasonable choice would consist in selecting as starting guess an existing configuration, so that the design could be improved. In this way an improvement should be guaranteed unless the initial design already is a local optimum. Other strategies require much more computational effort but try to deal with initial guess dependent problems. It is the case of multi-start strategies [Dixon and Szegő, 1975]. The design loop can start once the initial guess choice is made. This design is tested within FEA block so that displacement vectors can be computed. Afterward Post-Processing and Adjoint sensitivity are applied to compute the optimization problem responses and sensitivities. Finally the optimization solver provides a new design candidate. At this point convergence has to be tested. Actually in some variants, convergence may be tested before MMA Optimizer gives a new candidate. Optimization convergence criteria can be based on the objective function's variation over several optimization iterations, can be based on the variation of the design variable's vector between consecutive iterations, on the number of iterations or on the verification of first order optimality conditions (KKT conditions). If a solution is not attained, the iteration counter is increased and the new design candidate is sent to the FEA block. If convergence is attained the optimization loop is stopped and the design can be considered as a candidate solution. It is important to underline the fact that automated optimization loop cannot provide optimal design without human intervention. Often the designer needs to make some decision about:

- The fact that a design candidate is an actual local minimum. In fact often a maximum iteration stopping criterion is reached, without reaching the other criteria (change of objective or change in the design vector less than a given tolerance). In this case either one accepts the design even if it is not an optimum, either the iteration convergence criterion has to be increased. In other more delicate situations the final design is not feasible even after a long convergence history. This can be due to the fact that the optimization problem is said to be over-constrained (there are no feasible solutions), or can be due to a bad choice of the initial guess. In more rare cases the optimizer was not correctly tuned to deal with the problem in hand or a scaling has to be introduced in the problem formulation. MMA optimizer needs special care for the choice of some parameters that make the difference on the final outcome.
- The fact that a design candidate was correctly simulated. In other words if the design falls inside the model's validity domain. If this is the case one can either change the model's assumptions (for instance including nonlinearity in the model, changing the finite element order etc...) , either keep the same model but enforce the design to avoid such solutions changing both formulation and model.
- The fact that a design candidate is acceptable, i.e. is a manufacturable solution and is compliant with all requirements that can be identified on a quantifiable and qualitative level. This is a much more complex requirement for a design. Sometimes these manufacturing constraints may be considered during the optimization loop. It must be noticed that for this phase complex and expensive multidisciplinary simulations could be necessary to validate a design.

The complete design loop described above is shown in figure 2.13. One can observe that this procedure can be time consuming, but avoids the call to complex simulations that may not be required in the design loop as they can be verified at the end of each optimization loop. Practically it can happen that some responses are still too expensive to be considered inside the design loop. In this case surrogate based optimization can represent a valid solution [Forrester et al., 2008].

## 2.2 Density based topology optimization

In topology optimization, one seeks to achieve the design solution load path only knowing its design domain volume and its operating conditions. Since the pioneering work of Bendsoe and Kikuchi (1988) [Bendsøe and Kikuchi, 1988], topology optimization was developed and applied in several fields and physical problems. This thesis focused on SIMP (Solid Isotropic Material with Penalization) [Bendsøe, 1989]. This topology optimization approach is the most adopted in literature [Deaton and Grandhi, 2014] probably due to its simplicity and its robustness. Many other approaches could be considered as Level Set approach [Wang et al., 2003], Evolutionary Structural Optimization (ESO) [Xie and Steven, 1993] or biologically-inspired layout and topology optimization method [Kobayashi, 2010]. The interested reader

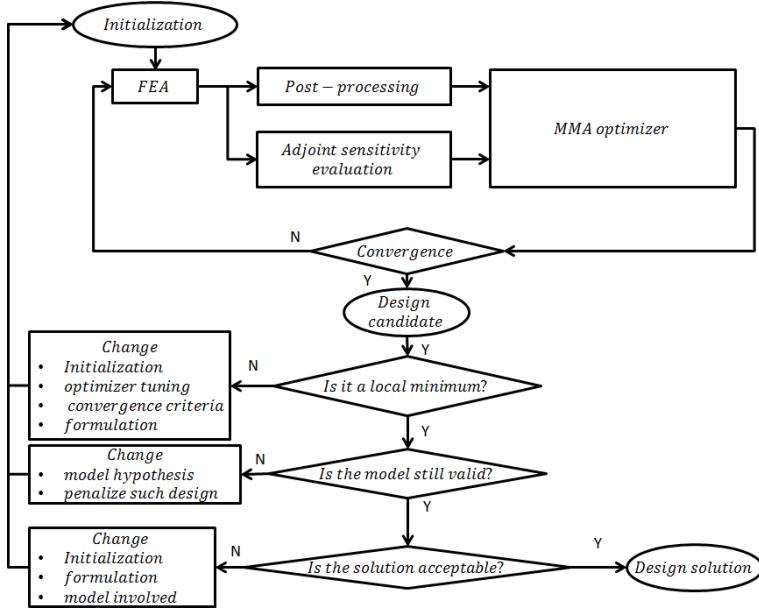


Figure 2.13: Design loop block diagram representation

can find a review in [Deaton and Grandhi, 2014]. All these approaches show some advantages and shortcomings with respect to SIMP approach. For this reason, as a first attempt, SIMP was implemented. Future research could actually benefit from the investigation of other Eulerian approaches but are out of the scope of this dissertation.

### 2.2.1 Problem formulation

In this method, the solution is described by a density field between zero and one that locally gives the information on the presence or the absence of material in terms of both mass and Young's modulus. A solution presenting intermediate density elements is always possible but is penalized by the power interpolation law used for the Young's modulus [Bendsøe, 1989]. Let us introduce a pseudo density field  $x(\{X_g\})$  in the design zone  $\Omega$ . This field has a physical meaning of presence or absence of material so that should take a value in  $\{0, 1\}$  reading 0 as absence of material and 1 as presence of material. For  $x(\{X_g\}) = 1$  one should use the value of local density and Young's Modulus of the real material  $E$ . In the case  $x(\{X_g\}) = 0$  the void can be simulated using a very soft material Young's Modulus  $E_{min}$  that prevents stiffness matrix singularity and ill-conditioning. In order to use efficient gradient based optimization algorithms, the problem is commonly relaxed so that  $x(\{X_g\}) \in [0, 1]$ . The physical interpretation of results presenting large regions characterized by intermediate density is not easy as they cannot be interpreted either as full material or void. For this reason, a simple penalization technique is commonly employed to prevent the optimization algorithm from converging on this kind of solutions. For this reason in the SIMP method [Bendsøe,

1989], the Young's modulus is written as a function of  $x$  as:

$$E(x) = E_{min} + (E - E_{min})x^p \quad (2.40)$$

Where the penalty value  $p > 1$ , penalizes the stiffness of intermediate densities. The value of  $p$  is usually set to 3 in order to get nearly black and white solutions. The Finite Element stiffness matrix assembly procedure will then use the local value of the Young's Modulus in order to compute the stiffness matrix. In the proposed framework the Young's modulus was considered as constant in each finite element so that the element stiffness matrix can be computed using the value of  $x$  in the element centroid  $x_{el}$  and the unit modulus element stiffness matrix  $[K_{el}^{(1)}]$ :

$$[K_{el}(x_{el})] = E(x_{el}) [K_{el}^{(1)}] \quad (2.41)$$

The design zone stiffness matrix is assembled summing the contribution of each element matrix in a classic FE style:

$$[K_{DZ}(\{x\})] = \bigoplus_{el=1}^{N_{el}} [K_{el}(x_{el})] \quad (2.42)$$

Where  $\bigoplus$  represents the assembly finite element operator and  $N_{el}$  is the number of elements in the design zone. Loads and boundary conditions are directly applied to the finite element model of the design zone. At this point for each material layout it is possible to compute the corresponding displacements vector. Historically topology optimization is associated with compliance minimization under volume constraints. This means that one looks for the stiffest material layout with respect to the loading conditions that fit inside the design zone within a given amount of material mass. Formally the following problem need to be solved<sup>7</sup>:

$$\begin{cases} \min_{\{0\} \leq \{x\} \leq \{1\}} \{U(\{x\})\}^T [K_{DZ}(\{x\})] \{U(\{x\})\} \\ s.t. \\ \frac{\{|\Omega_{el}|\}^T \{x\}}{\{|\Omega_{el}|\}^T \{1\}} - v_f \leq 0 \end{cases} \quad (2.43)$$

Where  $\{|\Omega_{el}|\}$  is the element area vector,  $\{1\}$  and  $\{0\}$  are vector with the same size of  $\{x\}$  whose values are all 1 and 0 respectively and  $v_f$  is the value of allowable volume fraction. Optimization algorithms like the Optimality Criteria [Bendsøe and Sigmund, 1995] or the aforementioned SCP algorithms [Svanberg, 1987] are often adopted for the solution of this nonlinear programming problem. Once a design is determined, topology optimization solutions need to be re-interpreted and transformed in a realistic design. At this phase some model assumptions could be considered and further analysis and optimization could be performed until a final design is reached. For this second phase it is important to understand a solution's mechanical behavior and its global load path. Such considerations could come into account and the final solution could therefore be sensibly different from the original design determined in the topology optimization phase.

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<sup>7</sup>Here only one load case was considered for brevity, the generalization to multiple load case will be detailed later.

### 2.2.2 Numerical difficulties & density filter

The optimization problem 2.43 suffers of several numerical difficulties [Sigmund and Petersson, 1998]:

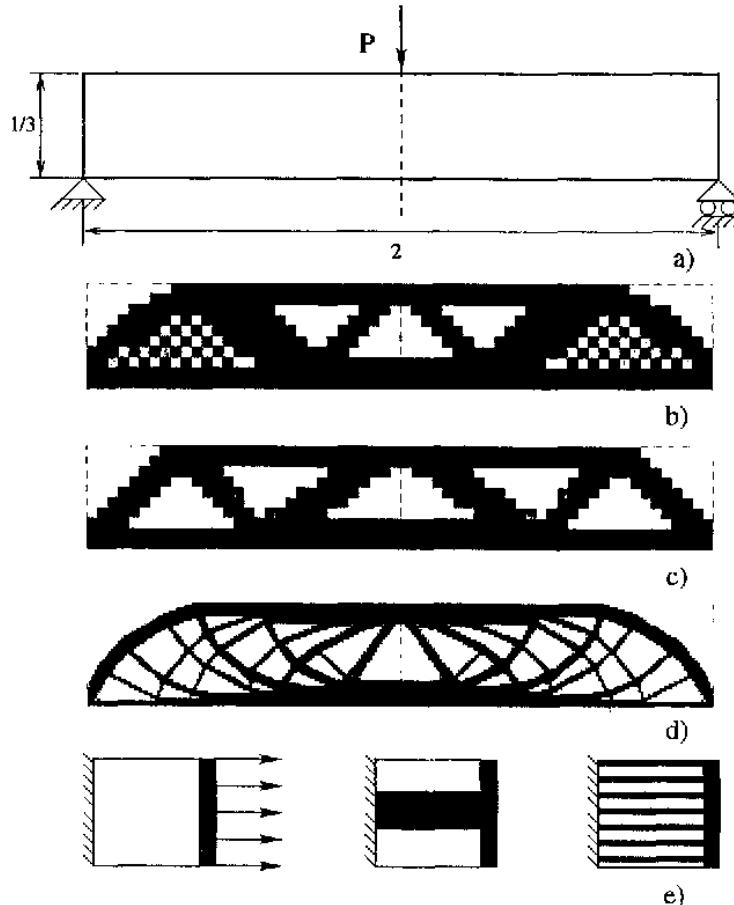


Figure 2.14: Numerical instabilities in topology optimization [Sigmund and Petersson, 1998],  
a) Problem definition b) solution with checkerboards, c) solution without checkerboards for a coarse mesh, d) solution with a refined mesh d) Example of problem with an infinity of local minima with the same stiffness and volume fraction

- *Mesh dependence:* The solution of the optimization problem depends on the discretization adopted for the FEM. Mesh dependence is linked to the fact that the problem originating discretized topology optimization (i.e. the continuum topology optimization problem with 0-1 density) doesn't have a solution (this problem is also known as nonexistence [Sigmund and Petersson, 1998]). Moreover the second issue related to this problem is the non-uniqueness of the solution. To deal with these difficulties several restriction techniques can be found in the literature. Here it is possible to cite the perimeter control method [Ambrosio and Buttazzo, 1993, Haber et al., 1996], the global gradient constraint [Bendsøe and Sigmund, 1995], the local gradient constraints [Niord-

son, 1983] and the mesh independent filtering [Sigmund, 1994, Sigmund, 1997] (for a larger review the reader can refer to [Sigmund, 2007]).

- *Checkerboards*: The solution of 2.43 shows zones of alternated full and void materials in a checkerboard like fashion. It was shown by [Jog and Haber, 1996] that this is due to FEM non convergence when representing such structures. Several solutions have been proposed to avoid such problems, for instance: Higher-order finite elements [Diaz and Sigmund, 1995, Jog and Haber, 1996], the Patches [Bendsøe et al., 1993] and filtering [Sigmund, 1994].
- *Local minima*: Changing algorithmic parameters one can find different solutions to the same problem, with the same FEM discretization. This is due to the optimization problem non convexity, and to the fact that optimizer adopted for topology optimization are most of the time local solvers. Global solvers are in fact less adapted to deal with such problems due to the large number of design variables. On the other hand continuation approaches can provide global or at least an improved solution. The idea is to start by solving a more convex problem (for instance for  $p = 1$ ) and then progressively move to the original non convex one (for instance progressively increasing the value of  $p$ ), altering some parameters (penalty, filter radius or others) in both model and optimization formulation . Among classic approaches one can find the works of [Allaire and Francfort, 1993, Allaire and Kohn, 1993], strategies based on the perimeter constraint [Haber et al., 1996], on the value of the filter radius [Sigmund, 1997, Sigmund and Torquato, 1997] or approaches based on the penalization of intermediate densities [Guedes and Taylor, 1997] .

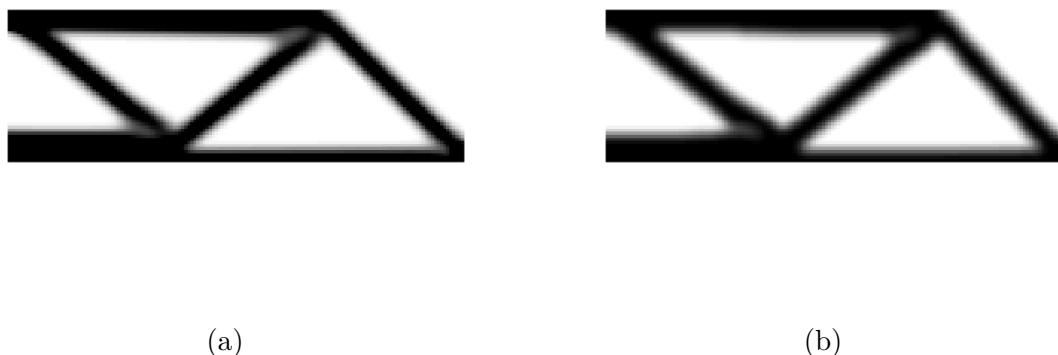


Figure 2.15: a) sensitivity filters and b) density filters effect on the solution of a  $150 \times 50$  mesh MBB problem [Andreassen et al., 2011].

In this thesis mesh independent density filters [Bourdin, 2001, Bruns and Tortorelli, 2001] were adopted. As the considered meshes are non-uniform, the procedure adopted for their computation is detailed here. Typically mesh dependency and checkerboard related issues are

solved using mesh independent filtering techniques. The density filter transforms the original density vector using an integral operator that in the continuous form is defined as:

$$x_{Phys}(\{X_g\}) = \int_{\Omega} \kappa(\{X_g\}, \{\bar{X}_g\}) x(\{\bar{X}_g\}) d\{\bar{X}_g\} \quad (2.44)$$

In this way even if the density distribution is rough, the filtered version defined by  $\{x_{Phys}(\{X_g\})\}$  will inherit the smoothness of the kernel functions  $\kappa(\{X_g\}, \{\bar{X}_g\})$ . The latter is often chosen to be a linear hat kernel with radius  $r$ :

$$\kappa(\{X_g\}, \{\bar{X}_g\}) = \hat{\kappa}(\{X_g\}) \max \left( 0, 1 - \frac{\|\{X_g\} - \{\bar{X}_g\}\|}{r} \right) \quad (2.45)$$

where:

$$\hat{\kappa}(\{X_g\}) = \left( \int_{\Omega} \max \left( 0, 1 - \frac{\|\{X_g\} - \{\bar{X}_g\}\|}{r} \right) d\{\bar{X}_g\} \right) \quad (2.46)$$

In a discretized version, these operations take the matrix form:

$$\{x_{Phys}\} = [H] \{x\} \quad (2.47)$$

Where the integral of equations (2.44), (2.46) where replaced by Gauss numerical integrations using finite elements as support. These techniques can be easily implemented for uniform structured meshes as is the case in [Andreassen et al., 2011] and the majority of SIMP based topology optimization studies. The fact of having 1x1 square element can be used to have a straightforward relationship between the element indexing and their neighbors center-to-center distances. This is not the case for non-uniform unstructured meshes, as the ones considered here. In Talischi et al. [Talischi et al., 2012] the difficulties induced by those cases are treated for 2D analysis with unstructured polygonal meshes. In particular it is shown that, to be more efficient, stiffness element matrices and filter matrix can be assembled once for all before the optimization loop. In this study, 8-node 3D finite elements with tri-linear shape functions were employed and 8 Gauss points per element were considered for stiffness matrix assembly. On the other hand only one Gauss point was used in order to evaluate the filtering convolution integral. As a consequence, following the same implementation of [Talischi et al., 2012] the filter matrix  $[H]$  needs the evaluation of all distances  $d_{ij}$  between each couple of element's centroid:

$$H_{i,j} = \frac{\max(0, |\Omega_j|(1 - \frac{d_{ij}}{r}))}{\sum_{j=1}^{N_{el}} \max(0, |\Omega_j|(1 - \frac{d_{ij}}{r}))} \quad (2.48)$$

Where  $N_{el}$  is the element number in the design zone,  $r$  is filter radius and  $|\Omega_j|$  is the volume of  $j^{th}$  element. The cost for this evaluation in terms of memory and CPUs time grows with  $\frac{N_{el}^2 - N_{el}}{2}$ . In [Talischi et al., 2012] it is argued that this operation even if expensive should be done once for all before the optimization loop and should not be a bottleneck for overall analysis. Nevertheless, when increasing the number of elements, this simple operation can encounter memory limits faster than stiffness matrix inversion. These limitations were also studied and tackled in the PDE filter proposed by Lazarov et al [Lazarov and Sigmund, 2011],

where instead of explicitly computing the filter matrix  $[H]$ , the filtered field is found as the solution of a PDE problem. Such a solution involves a small memory cost, even if it requires the solution of a system of equations with the size the number of nodes, twice per iteration. These solutions are not very expensive for small problems. In fact they require a small effort compared to displacements evaluation. In this work an alternative way of directly computing  $[H]$  was proposed. This reduces the time needed for the computation of the filtering matrix, but still requires enough memory for the storage of  $[H]$ . Note that for reasonably small values of the filtering radius, and for unstructured refined meshes, this proposed procedure appears advantageous. On the other hand for large values of the filtering radius and for the same kind of meshes the approach based on the PDE filter by Lazarov et al. [Lazarov and Sigmund, 2011] is to be preferred to reduce memory requirements. It must be noted that  $H_{i,j}$  is sparse since all distances superior to  $r$  do not contribute to  $H_{i,j}$ . If one suppose to have a first mesh like the one in figure 2.16a on which it is possible to evaluate all the distances between each element  $d_{ij}$  and compare them to  $r$ . A refinement of this mesh can be obtained cutting each element into 8 as shown in figure 2.17a. Let's consider two coarser mesh elements and their

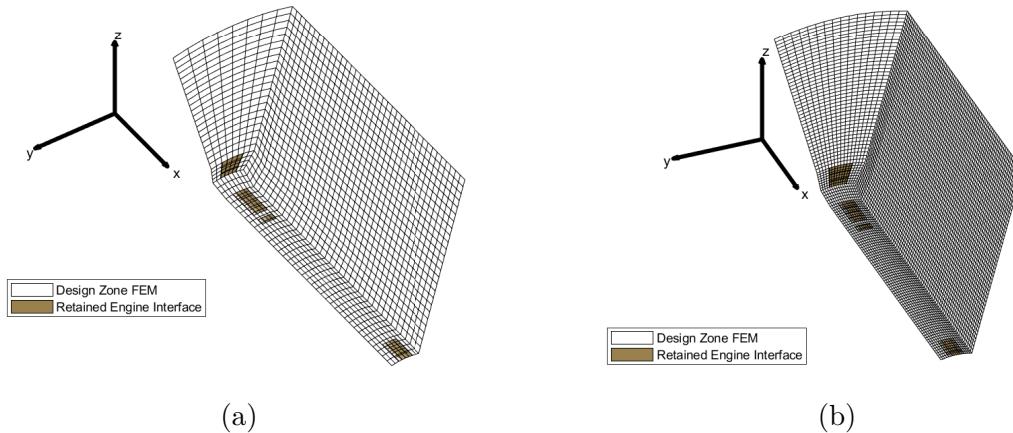


Figure 2.16: Mesh refinement procedure needed for multi-grid approach for the evaluation of  $[H]$ , the engine reduced element set are colored in yellow. Each node of these elements is kept in the engine super element.(a) Original design zone mesh, generated in Abaqus and imported in Matlab by input file parsing. The mesh counts 7600 8-node linear 3D solid finite elements and 9126 nodes. (b) Design zone mesh after refinement. Each element of the original element is cut in 8 new elements, that give a total of  $7600 \times 8 = 60800$  8-node linear 3D solid finite elements and 66759 nodes.

partition as considered in figure 2.17b whose centroid distance is known  $\|\vec{AC}\|$ . The minimal distance between centroids of the corresponding finer mesh elements,  $\|\vec{BD}\|$  can be related to  $\|\vec{AC}\|$  as :

$$\|\vec{AC}\| \leq \|\vec{AB}\| + \|\vec{BD}\| + |\vec{CD}| \quad (2.49)$$

So that:

$$\|\vec{BD}\| \geq \|\vec{AC}\| - \|\vec{AB}\| - |\vec{CD}| \quad (2.50)$$

The distances  $\|\vec{AB}\|$  and  $\|\vec{CD}\|$  are also bounded by the radius of the smallest sphere circumscribed around the biggest coarser mesh element  $r_c$ . Therefore equation (2.50) becomes:

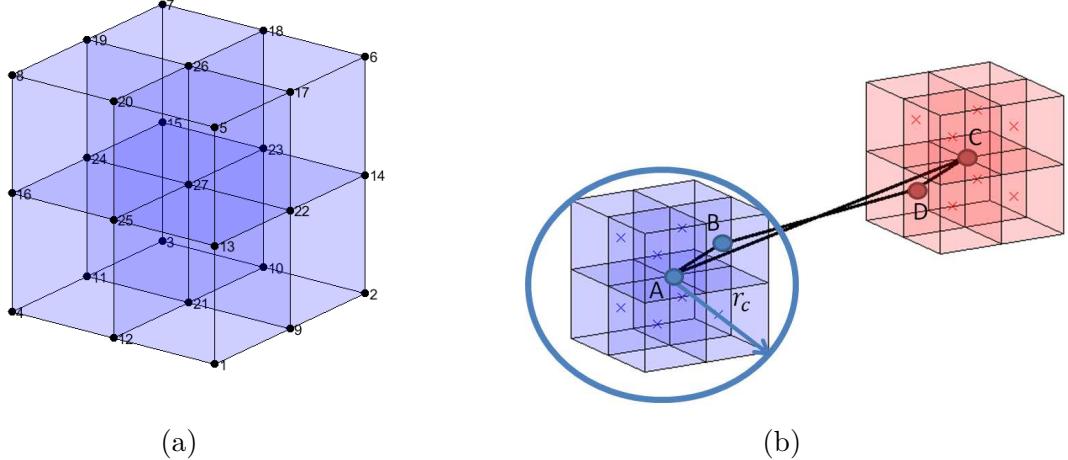


Figure 2.17: Relation between coarse and fine mesh. (a) 3D partition employed for mesh refinement, nodes 1 to 8 belong to the original coarse mesh element. Partition determines node 9 to 27 and 8 elements of the finer mesh. (b) Scheme of two elements of the original mesh after partition. The relationship between the minimal distance between two finite element centroids in the finer mesh  $\|\vec{BD}\|$ , the corresponding distance between the centroids of the coarse mesh  $\|\vec{AC}\|$  and the radius of the sphere circumscribed around the biggest coarse mesh finite element  $r_c$  can be determined considering the vector chain  $\vec{AC} = \vec{AB} + \vec{BD} + \vec{DC}$ .

$$\|\vec{BD}\| \geq \|\vec{AC}\| - 2r_c \quad (2.51)$$

Note that even if the scheme of figure 2.17b considers cubic elements equation 2.51 is valid for general 8-node brick elements. If the distances between coarse mesh centroids are known, it is possible to set-up a test on these distances that can help to reduce the number of centroid-to-centroid distances that have to be computed for the finer mesh. In fact for equation 2.51:

$$\|\vec{AC}\| \geq 2r_c + r \Rightarrow \|\vec{BD}\| \geq r \quad (2.52)$$

Then the distance that for sure needs not to be computed in the finer mesh are the one between elements obtained from coarser mesh at a distance greater than  $2r_c + r$ . On the other hand it is not possible to conclude that each and every distance that one can evaluate in this way will be smaller than  $r$ . The final cost of  $[H]$  is then equal to  $28N_c + 64N_c^*$  where  $N_c$  is the number of elements in the coarse mesh and  $N_c^*$  is the number of coarse mesh element pairs at a distance less or equal to  $2r_c + r$ . Since  $N_c = \frac{N_{el}}{8}$  and for reasonably small  $r$ ,  $N_c^* = KN_c \ll (N_c)^2$  the cost for this procedure grows up with  $(\frac{7}{4} + 8K)N_{el}$ , linearly and not quadratically with the problem size  $N_{el}$ . One can note that this procedure is also suitable for parallel implementations, thus further decreasing its numerical cost.

### 2.2.3 2D topology optimization examples

This section presents the implementation of 2D structural topology optimization of structures that are supposed to be in plane stress. The implementation stems from the very well-known 88 lines Matlab code [[Andreassen et al., 2011](#)], that here has been implemented with some modifications:

- The plots have been changed to include load and boundary condition application as the convergence history of compliance and volume fraction.
- The test cases considered are the MBB beam, the Short cantilever beam and the L-shape design. The next section will only consider L-shape topology optimization, the other case are in appendix B.1.
- Heaviside filter [[Guest et al., 2004](#), [Lazarov and Sigmund, 2011](#)], penalty [[Allaire and Francfort, 1993](#), [Allaire and Kohn, 1993](#)], and filter radius [[Sigmund, 1997](#), [Sigmund and Torquato, 1997](#)] continuation were implemented to enforce the solution toward improved local optima with smaller gray regions. In particular SIMP penalty was increased from 1 to 6, increasing the value of the penalty of 1 each 75 iterations or when the convergence is attained. Similarly the  $\beta$  parameter of the Heaviside filter was increased from 1 to 10. Finally the value of the filter radius was decreased from a starting value to a minimum value of 2 decreasing its value of 1 every 75 iterations or when convergence is attained.
- The Optimality Criteria solver was replaced by MMA solver.

Table 2.1: Parameters for 2D topology optimization

Parameter	symbol	value
Full material Young modulus	$E_0$	1
Void material Young modulus	$E_{min}$	$10^{-9}$
Poisson ratio	$\nu$	0.3
Force amplitude	$F$	1
Asymptotes initialization	asyinit	0.5
Moving limit	move	0.1
Filter type	ft	3
Volume fraction	$\nu$	0.3
Initial guess	$\{x_0\}$	$0.3\{1\}$
Stopping criteria change	tol	$10^{-2}$
Stopping criteria iterations	maxiter	1000

The parameters selected for the following analysis are presented in table 2.1 <sup>8</sup>. The filter type is equal to 3 and this means that Heaviside filter is adopted.

<sup>8</sup>the value chosen for the Young's modulus are equal to the one considered by default in top88 Matlab code [[Andreassen et al., 2011](#)].

### 2.2.3.1 L-shape optimization

An interesting 2D use-case is the L-shape optimization c.f. figure 2.18. From an implemen-

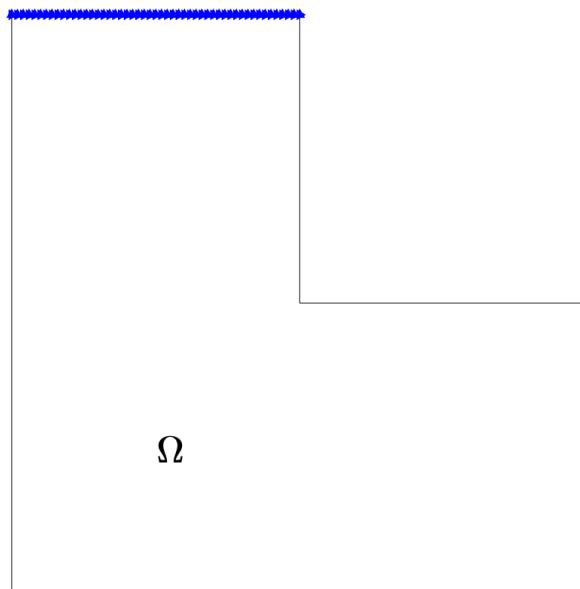


Figure 2.18: Geometry, Load and boundary conditions of the L-shape topology optimization problem. Blue triangles are oriented as the fixed DOFs. The red arrows represent the applied loads.

tation point of view this can be treated with the same top88 framework simply including passive elements. Another important reason that makes this use-case particularly interesting is the fact that the geometry includes at the inner corner a point of stress singularity in linear elasticity. For this reason it will be worth to mention this use-case again when speaking about stress based topology optimization. Here 2 discretizations are studied: a  $50 \times 50$  and a  $100 \times 100$  mesh. The initial value of the filter radius was set for the first mesh equal to 4 and for the second mesh equal to 8 (here it is recalled that the element side length is always 1 in this case). The results are shown in figure 2.19. A first important observation is that results are not similar. In fact in the design obtained for the coarser mesh it is possible to identify 13 thin members against 11 for the solution found by the refined mesh. This is due to the fact that even using continuation and density filters, the optimization problem still is far from being convex. For this reason finding different local minima is always possible. On the other hand both designs are nearly black and white, both are converged within the maximum iteration number. Selecting the design with the smallest number of members for ease of realization one can still be prone to refuse such design. In fact it is clear that both these designs will have very high stresses in the inner corner and will not be able to bear the load applied to the structure without encountering at least local plasticity, or even failure. In this case it is possible to say that the model is no longer valid for the solution proposed, and

that the formulation cannot be considered as acceptable for such a problem.

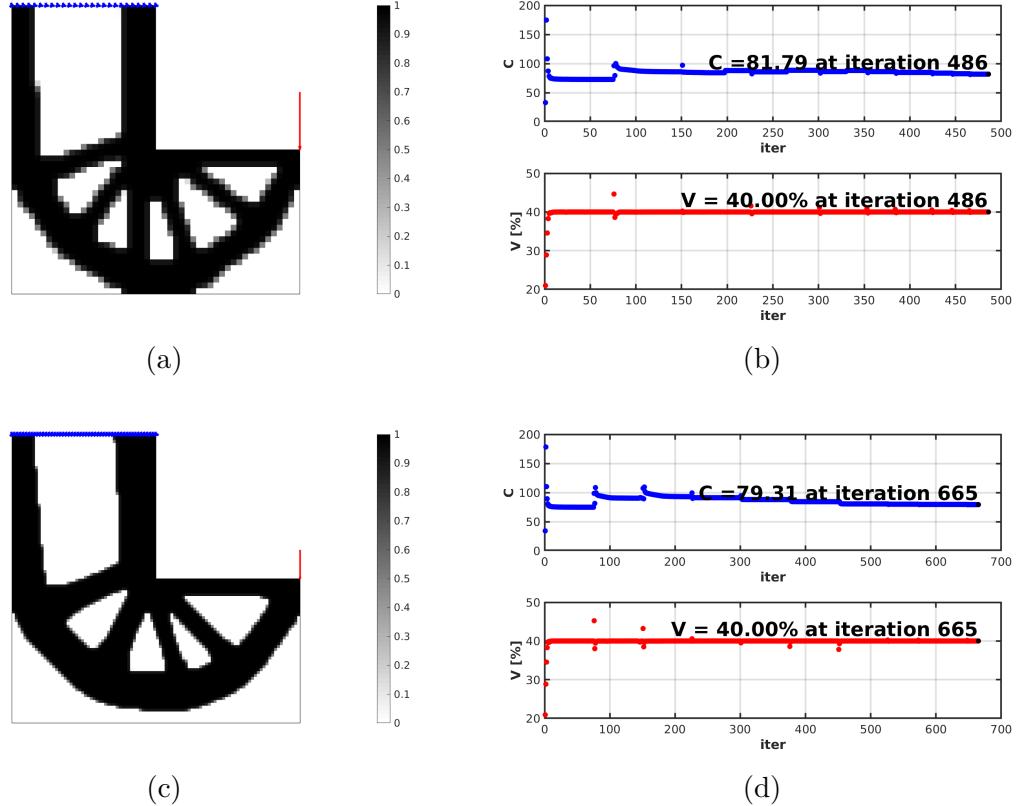


Figure 2.19: L-shape topology optimization results. (a)-(b)  $\{x_{Phys}\}$  distribution and responses convergence for the  $50 \times 50$  discretization.(c)-(d)  $\{x_{Phys}\}$  distribution and responses convergence for the  $100 \times 100$  discretization

## 2.3 Stress based topology optimization

Topology optimization is often related to the minimum compliance problem as it was discussed in the previous section. The problem of this kind of responses is the need of numerous changes that are required in a topology optimization solution to respect failure requirements. For metallic materials a common failure criterion adopted is the von Mises stress that in plane stress can be computed as:

$$\sigma_{VM} = \sqrt{\sigma_{xx}^2 + \sigma_{yy}^2 - \sigma_{xx}\sigma_{yy} + 3\tau_{xy}^2} \quad (2.53)$$

where  $\sigma_{xx}$ ,  $\sigma_{yy}$ ,  $\tau_{xy}$  are the corresponding components of the stress tensor. In the FEA they can be computed from the nodal displacements as in equation (1.40). In bilinear finite elements like the ones adopted in top88 code, the stresses are not constant over each finite element. If a single query point is considered in the middle of the element, matrices  $[D]$  and  $[B]$  of equation (1.40) becomes:

$$[B] = \frac{1}{4a} \begin{bmatrix} -1 & 0 & 1 & 0 & 1 & 0 & -1 & 0 \\ 0 & -1 & 0 & -1 & 0 & 1 & 0 & 1 \\ -1 & -1 & -1 & 1 & 1 & 1 & 1 & -1 \end{bmatrix} \quad (2.54)$$

$$[D] = \frac{E}{1-v^2} \begin{bmatrix} 1 & v & 0 \\ v & 1 & 0 \\ 0 & 0 & \frac{1-v}{2} \end{bmatrix} \quad (2.55)$$

where  $2a$  is the mesh side length. For the top88 framework  $a = \frac{1}{2}$ . In figure 2.20 the von Mises stress was shown in each finite element of the solution of figure 2.19c. As aforementioned 2 stress singularities can be found in both the load application region and in the L-shape inner corner. The first depends on the stress model applicability when loads are modeled as nodal forces. This singularity can be eliminated simply changing the modeling hypothesis and considering a distributed load instead. The second is an actual singularity. In fact in the inner corner local plasticity will be attained if the design is manufactured as the solution. To avoid such issues, a fillet should be considered and the solution should therefore be modified. Moreover it is possible to say that the problem formulation 2.43 doesn't enforce the solution to respect any allowable limit on the von Mises. Including stress constraints directly inside the design loop is therefore useful because it can help reducing the gap between optimized and final design. Instead of focusing on the stiffest design respecting some mass requirement, it is often more reasonable to think about the lightest design that fulfills all failure criteria i.e.:

$$\begin{cases} \min_{\{0\} \leq \{x\} \leq \{1\}} \{|\Omega_{el}|\}^T \{x\} \\ s.t. \\ (\sigma_{VM})_i \leq \sigma_{lim} \quad \forall i | x_i > 0 \end{cases} \quad (2.56)$$

On the other hand several difficulties arise when including such responses in the optimization formulation. In this section is about stress based topology optimization and about the challenges that need to be addressed due to the nature of stress constraints. Two major challenges arise in the implementation of stress constraints: the fact that the optimization

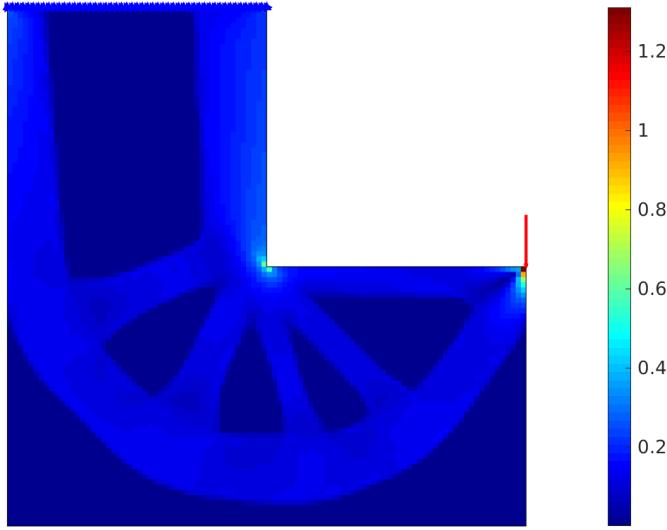


Figure 2.20: von Mises stress map for the solution of the L-shape optimization for the  $100 \times 100$  mesh.

problem presents singular optima [Kirsch, 1990, Cheng and Guo, 1997, Rozvany, 2001], and the large number of stress constraints to be considered. The first issue consists in the fact that local optima belong to degenerate subspaces of the feasible domain that are not reachable using standard gradient-based optimizations. The second issue consists in the fact that stress constraints are local, i.e. the stress should be controlled in every point of the structure. Theoretically an infinity of constraints should therefore be considered in the optimization formulation. Instead the stress is limited only in each finite element Gauss point (where the stress recovered by finite element method is the most accurate [Zlámal, 1978, Zhang and Naga, 2006]). Still the number of constraints to be considered is very high and most optimizers available have very hard time dealing with so many local constraints. One way of dealing with singular optima consists in relaxing stress constraints (for instance using  $\epsilon$ -relaxation [Cheng and Guo, 1997], or the q-p approach [Bruggi, 2008]). To avoid the consideration of many local constraints to the optimization formulation a global constraint can be considered instead, using the maximum function. To avoid derivation difficulties, regular approximation of the maximum function (for instance the Kreisselmeier-Steinhauser function [Kreisselmeier and Steinhauser, 1980, Yang and Chen, 1996] or the p-norm [Duysinx and Sigmund, 1998] ) can be considered. Aggregation techniques have the main drawbacks of not having a precise control on the final design maximum stress. To control the exact value of maximum stress in the final solution Le et al. [Le et al., 2010] proposed an adaptive approach that helps obtaining designs with the desired maximum stress. In this work the formulation presented in [Verbart et al., 2017] was adopted. In this approach each finite element is considered as a composite containing both full material and voids. The collapse in the structure is considered attained when the stress inside the full material reaches the allowable. For this reason one

has to describe the microscopic stress inside a finite element as:

$$\hat{\sigma}_{VM} = \frac{E_0 \sigma_{VM}}{E} \quad (2.57)$$

In this way the stress computed in element that have densities approaching zero, will tend to a finite value and not to 0. Then the relaxed constraint violations can be defined for each Gauss point in each finite element as:

$$\bar{g}_i = (x_{Phys})_i \left( \frac{(\hat{\sigma}_{VM})_i}{\sigma_{lim}} - 1 \right) \quad \forall i = 1, 2, \dots, N_{GP} \quad (2.58)$$

where  $(x_{Phys})_i$  is the physical density corresponding to the element including the  $i^{th}$  Gauss point. These violations are positive when the microscopic stresses are greater than the allowable but are equal to 0 when an element has zero density. In this way it is proven that singular optima can be attained by gradient based solver. The aggregated stress is finally evaluated by the use of the lower bound Kreisselmeier-Steinhauser function [Kreisselmeier and Steinhauer, 1980] that approximates the local relaxed stress constraint violation maximum:

$$G_{KS}^l = \frac{1}{P} \ln \left( \frac{1}{N_G} \sum_{i=1}^{N_G} e^{P\bar{g}_i} \right) \quad (2.59)$$

where  $N_G$  is the total number of Gauss point and  $P > 1$  is the aggregation constant. When evaluating equation (2.59) on design that violating the stress constraint, very large numbers are summed together with smaller numbers. This can lead to have a numerical infinity to be attained by the KS function due to the machine representation limitation. To avoid such numerical issues, the formulation of equation (2.59) can be rewritten as:

$$G_{KS}^l = \bar{g}_{max} + \frac{1}{P} \ln \left( \sum_{i=1}^{N_G} e^{P(\bar{g}_i - \bar{g}_{max})} \right) - \frac{\ln(N_G)}{P} \quad (2.60)$$

Where  $\bar{g}_{max} = \max_i \bar{g}_i$ . In Appendix B.2 some useful properties of the  $G_{KS}^l$  function are provided and equations (2.60) and (2.59) are proved to be equivalent. The reader can observe that equation (2.60) is differentiable and its derivatives is given by differentiation of equation (2.59) that is provided later in this section. The satisfaction of stress constraints is thus imposed in problem (2.56) as:

$$\begin{cases} \min_{\{0\} \leq \{x\} \leq \{1\}} \{|\Omega_{el}|\}^T \{x\} \\ s.t. \\ G_{KS}^l \leq 0 \end{cases} \quad (2.61)$$

The reader can observe that this is only one of the possible ways of aggregating constraints available in literature. One can also use P-mean [Duysinx and Sigmund, 1998] or Induced Exponential [Kennedy and Hicken, 2015] approximation of maximum. In chapter 3 these approaches will be described in much deeper details. A second important observation about formulation 2.61 is that  $G_{KS}^l$  need adjoint evaluation of sensitivities. Here a short computation

chain for the partial derivatives  $\left\{ \frac{\partial G_{KS}^l}{\partial x_{Phys}} \right\}$  and  $\left\{ \frac{\partial G_{KS}^l}{\partial U} \right\}$  is proposed:

$$\left\{ \frac{\partial G_{KS}^l}{\partial x_{Phys}} \right\}^T = \left\{ \frac{\partial G_{KS}^l}{\partial \bar{g}} \right\}^T \left[ \frac{\partial \bar{g}}{\partial x_{Phys}} \right] \quad (2.62)$$

$$\left\{ \frac{\partial G_{KS}^l}{\partial U} \right\}^T = \left\{ \frac{\partial G_{KS}^l}{\partial \bar{g}} \right\}^T \left[ \frac{\partial \bar{g}}{\partial U} \right] \quad (2.63)$$

where:

$$\frac{\partial G_{KS}^l}{\partial \bar{g}_i} = \frac{e^{P\bar{g}_i}}{\sum_{j=1}^{N_G} e^{P\bar{g}_j}} \quad (2.64)$$

$$\frac{\partial \bar{g}_i}{\partial x_j} = \delta_{ij} \left( \frac{(\hat{\sigma}_{VM})_i}{\sigma_{lim}} - 1 \right) \quad (2.65)$$

$$\frac{\partial \bar{g}_i}{\partial U_j} = \frac{(x_{Phys})_i}{\sigma_{lim}} \frac{\partial (\hat{\sigma}_{VM})_i}{\partial U_j} \quad (2.66)$$

$$\delta_{ij} = \begin{cases} 1 & \text{if } i \in GP_j \\ 0 & \text{otherwise.} \end{cases} \quad (2.67)$$

Where  $GP_j$  is the ensemble of j-th element Gauss Point indexes. Numerical issues similar to the one described for the evaluation of equation (2.59) are encountered in the evaluation of equation (2.64). Since the numerator and the denominator can be described as an infinite value by the machine a Not a Number (NaN) can be produced from this expression. This is why the following expression (equivalent to (2.64)) should be employed:

$$\frac{\partial G_{KS}^l}{\partial \bar{g}_i} = \frac{e^{P(\bar{g}_i - \bar{g}_{max})}}{\sum_{j=1}^{N_G} e^{P(\bar{g}_j - \bar{g}_{max})}} \quad (2.68)$$

For the computation of  $\frac{\partial(\hat{\sigma}_{VM})_i}{\partial U_j}$  it is worthy to introduce some simplifications in the expression of  $(\hat{\sigma}_{VM})_i$ :

$$(\hat{\sigma}_{VM})_i = \sqrt{\{\hat{\sigma}_i\}^T \begin{bmatrix} 1 & -0.5 & 0 \\ -0.5 & 1 & 0 \\ 0 & 0 & 3 \end{bmatrix} \{\hat{\sigma}_i\}} = \sqrt{\{U_i\}^T [\hat{S}] \{U_i\}} \quad (2.69)$$

$$\{\hat{\sigma}_i\} = \begin{Bmatrix} (\hat{\sigma}_{xx})_i \\ (\hat{\sigma}_{yy})_i \\ (\hat{\tau}_{xy})_i \end{Bmatrix} \quad (2.70)$$

Where  $\{\hat{\sigma}_i\}$  is the vector containing the stress components in the  $i^{th}$ -Gauss point,  $\{U_i\}$  represent the vector of nodal displacement of the element containing the same Gauss point taken

in the order of the local DOF sorting. Moreover:

$$[\hat{S}] = [B]^T [\hat{D}]^T [T] [\hat{D}] [B] \quad (2.71)$$

$$[T] = \begin{bmatrix} 1 & -0.5 & 0 \\ -0.5 & 1 & 0 \\ 0 & 0 & 3 \end{bmatrix} \quad (2.72)$$

$$[\hat{D}] = \frac{E_0}{E} [D] \quad (2.73)$$

where  $[T]$  is a constant matrix. It can be noticed that  $[\hat{D}]$  is a property of the full material and is not function of the physical density. With all these simplification considered:

$$\frac{\partial(\hat{\sigma}_{VM})_i}{\partial U_j} = \frac{\left\{ \frac{\partial\{U_i\}}{\partial U_j} \right\}^T [\hat{S}] \{U_i\}}{(\hat{\sigma}_{VM})_i} \quad (2.74)$$

where  $\left\{ \frac{\partial\{U_i\}}{\partial U_j} \right\}$  contains either 1 or 0 if the index  $j$  correspond to one of the component of vector  $\{U_i\}$ .

### 2.3.1 A 2D stress based topology optimization example

In this section the stress based formulation of eq. 2.61 is tested on the L-shape topology optimization problem. The stress based formulation is much more dependent on optimization tuning than the compliance one. To avoid MMA divergence, the optimizer step length (i.e. the distance between a design and the one at the previous iteration) need to be controlled. Several strategies can be adopted for this purpose directly using the MMA Matlab code [Svanberg, 1987]. One can use the move limit defined by the variable move. This variable controls the lower and upper bound defined in the approximated optimization problem, so that the optimization step length is directly controlled. There is a second way of reducing the optimizer step in a more indirect way. This consists in reducing the maximum distance of asymptotes from the current point. In this way the local approximation adopted for constraints will be violated faster. This implicitly reduces the optimizer step length when the sub-solver iterations converge to a minimum of the local approximation. As suggested by [Verbart et al., 2017] this second strategy was adopted. The domain of interest is  $100 \times 100$ . Both the effect of discretization and aggregation was considered. Continuation strategy was adapted as for the compliance formulation. On the other hand, this time the filter radius was kept constant to the value of 2 mesh size and the beta parameter was increased only up to 6. The increment of  $\beta$  and the SIMP penalty were applied every time that convergence was attained or every 300 iterations. The convergence criterion was also on the infinity norm of the variation of design vector. This time a value of 0.001 was selected as convergence tolerance. The maximum of iterations was set to 10000. To avoid modeling issues on the stress the nodal force applied as in figure 2.20 was replaced by a linear force distribution over a length of 5. In table 2.2<sup>9</sup> the

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<sup>9</sup>The values adopted in this study are similar to the one considered in [Verbart et al., 2017].

optimization set-up is summarized. The effect of  $P$  on the topology optimization is shown in

Table 2.2: Parameters for 2D stress based topology optimization

Parameter	symbol	value
Full material Young modulus	$E_0$	1
Void material Young modulus	$E_{min}$	$10^{-9}$
Full material allowable stress	$\sigma_{lim}$	1
Poisson ratio	$\nu$	0.3
pressure	$p$	$\frac{1}{5}$
Asymptotes initialization	asyinit	0.5
Moving limit	move	0.5
External Moving limit [Verbart et al., 2017]		0.1
Filter type	ft	3
Volume fraction	$v$	0.3
Initial guess	$\{x_0\}$	$\{1\}$
Stopping criteria change	tol	$10^{-3}$
Stopping criteria iterations	maxiter	10000

figure 2.21. A first important remark is that  $G_{KS}^l$  is always lower than the actual  $\bar{g}_{max}$ . As a consequence the maximal von Mises stress is always greater than the actual allowable of 1. Increasing the value of  $P$  the final stress is decreased, and this is a consequence of equation B.3. On the other hand MMA struggles to converge for increasing value of  $P$  as it can be seen on the constraint violation convergence history. Moreover even adopting a continuation strategy it can be observed that the solution presents some region of intermediate densities. Finally for  $P = 16$  and  $P = 28$  the sharp inner corner is replaced by a fillet. This a very well established design technique to avoid stress concentration which is also found by the optimization algorithm. To improve the accuracy of the maximum stress control one can increase  $P$  but this comes at the cost of a higher stress response non-linearity. In figure 2.23 the effect of  $\sigma_{lim}$  is shown for the same mesh discretization. It is interesting to notice that, for  $\sigma_{lim} = 0.9$  the final stress is actually  $\sigma_{max} \approx 1.05$  and that for  $\sigma_{lim} = 0.8$  the final stress becomes  $\sigma_{max} \approx 0.94$ . This suggests the possibility of ensuring the convergence to the results that possess a  $\sigma_{max} \approx 1$  by changing the value  $\sigma_{lim}$ . Such an approach is considered in section 3.9.

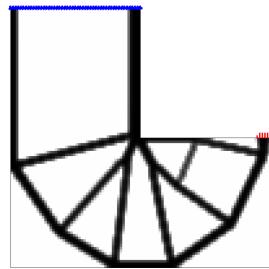
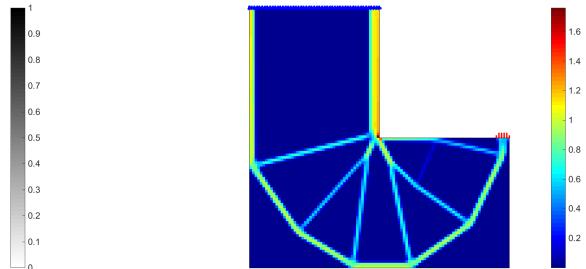
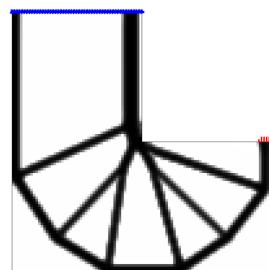
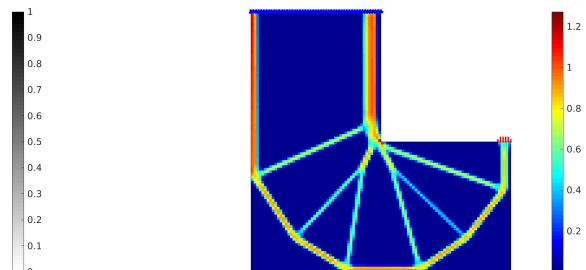
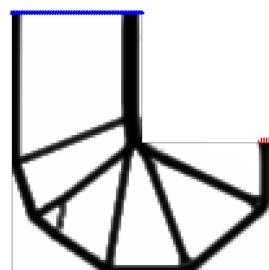
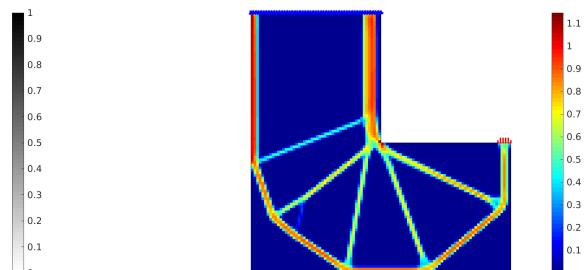
(a) Density plot  $P = 4$ (b) von Mises plot  $P = 4$ (c) Density plot  $P = 16$ (d) von Mises plot  $P = 16$ (e) Density plot  $P = 28$ (f) von Mises plot  $P = 28$ 

Figure 2.21: L-shape  $100 \times 100$  mesh resolution, stress based topology optimization results. Density distribution, von Mises stress distribution for  $P = 4, 16, 28$  and for  $\sigma_{lim} = 1$ .

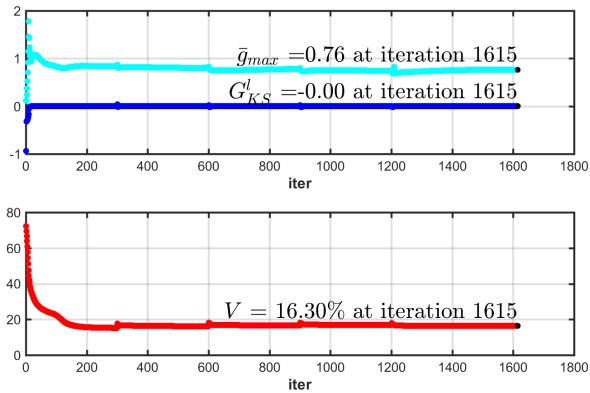
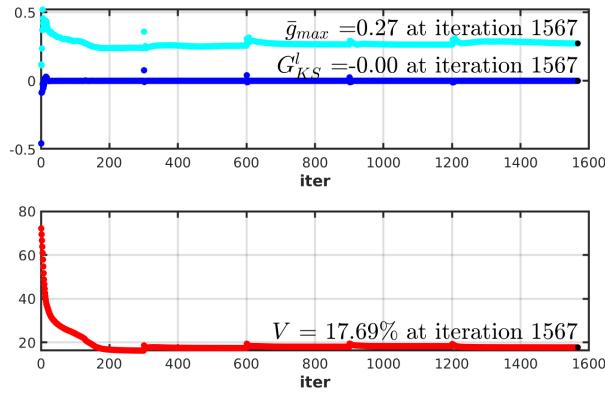
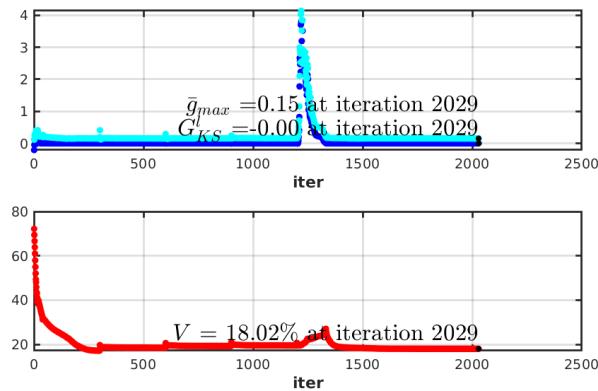
(a) Convergence plot  $P = 4$ (b) Convergence plot  $P = 16$ (c) Convergence plot  $P = 28$ 

Figure 2.22: L-shape  $100 \times 100$  mesh resolution, stress based topology optimization results. Convergence history for  $P = 4, 16, 28$  and for  $\sigma_{lim} = 1$ .

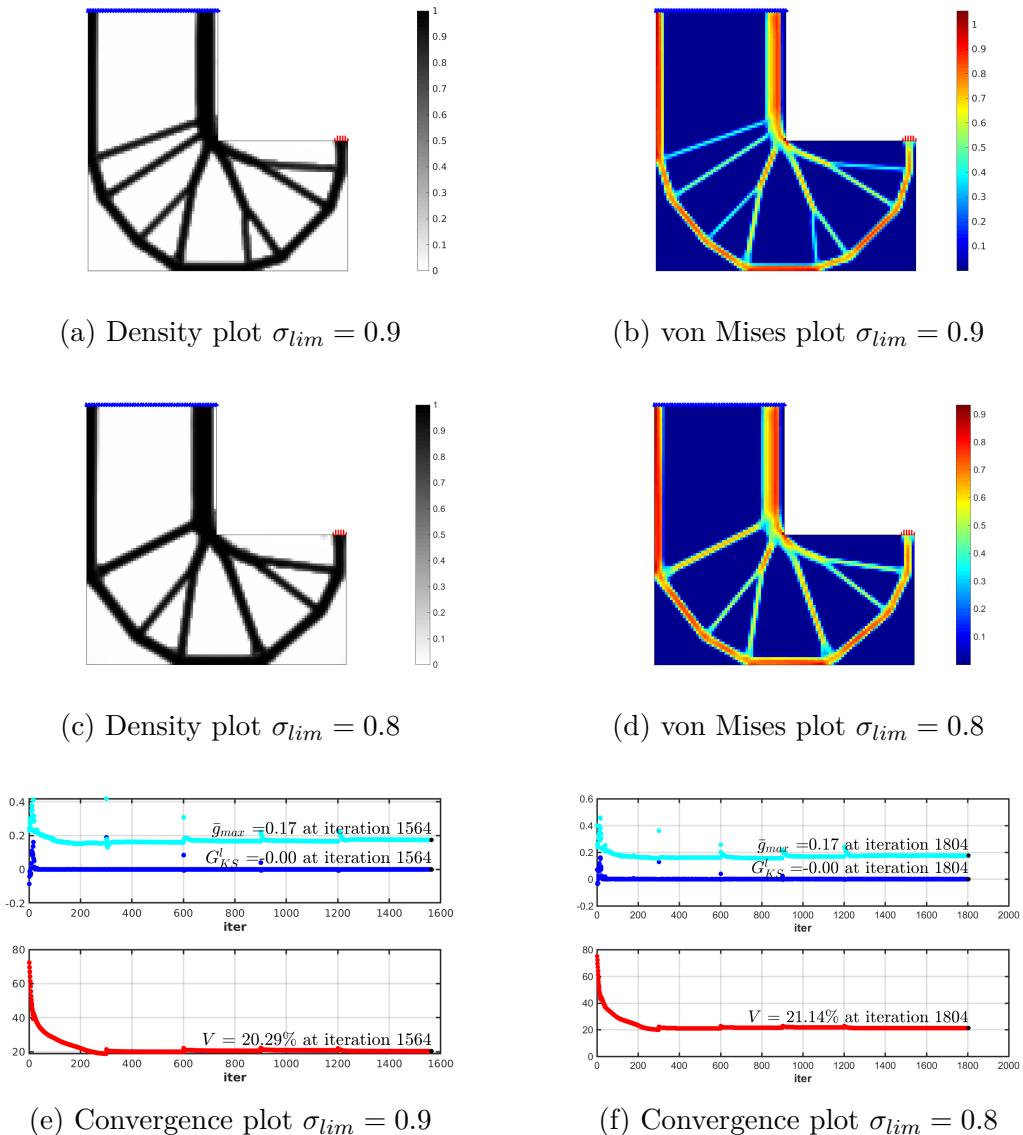


Figure 2.23: L-shape 100 × 100 mesh resolution, stress based topology optimization results. Density distribution, von Mises stress distribution and convergence history for  $P = 28$  and for  $\sigma_{lim} = 0.8, 0.9$ .

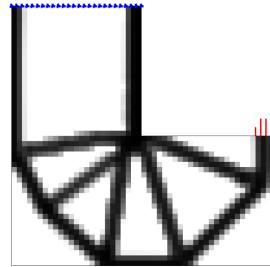
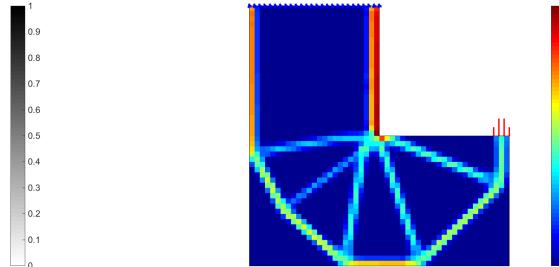
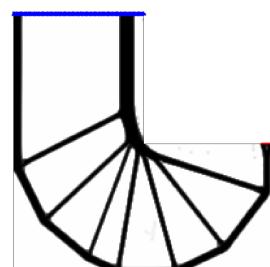
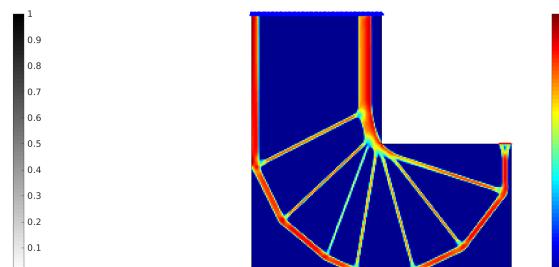
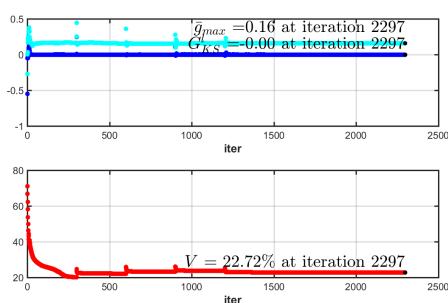
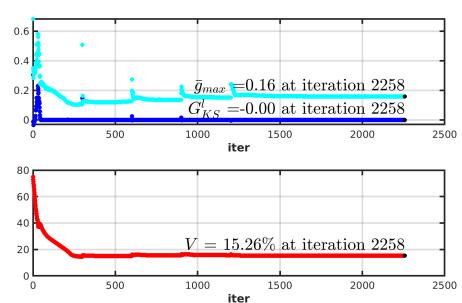
(a) Density plot  $50 \times 50$ (b) von Mises plot  $50 \times 50$ (c) Density plot  $200 \times 200$ (d) von Mises plot  $200 \times 200$ (e) Convergence plot  $50 \times 50$ (f) Convergence plot  $200 \times 200$ 

Figure 2.24: L-shape  $50 \times 50$  and  $200 \times 200$  mesh resolutions, stress based topology optimization results. Density distribution, von Mises stress distribution and convergence history for  $P = 28$  and for  $\sigma_{lim} = 1$ .

Finally in figure 2.24 the effect of mesh resolution is represented. The solutions are different for different meshes. This may be caused again by the optimization problem non convexity and more probably by the FEM convergence for the stress analysis. The final maximum stress seems to be not affected by the number of elements in the discretization. This may seem to not be in accordance with equation B.3. In reality this is due to the FEM improved accuracy, to the fact that different local minima are attained and to the fact that for finer meshes the stress peak could be attained in several Gauss point, attenuating the increase of distance from the actual  $\bar{g}_{max}$  that was expected due to equation B.3.

### 2.3.2 Dealing with multiple load cases

In many practical situations, a component will have to respect mechanical requirements in several operating conditions. When this is the case the topology optimization formulation needs to be modified to include such requirements simultaneously. In fact the best design for a given operating condition, can generally speaking be different from the best design with respect to another condition. Respecting the stress constraint for a loading condition will not prevent the structure from failing for another one. In the stress formulation 2.56, the von Mises stress should be controlled for all Gauss point but also for all operating conditions. Unified approach 2.61 is adopted for each condition giving the following nonlinear programming problem:

$$\begin{cases} \min_{\{0\} \leq \{x\} \leq \{1\}} \{|\Omega_{el}|\}^T \{x\} \\ s.t. \\ (G_{KS}^l)_{l_c} \leq 0 & \forall l_c = 1, 2, \dots, N_l \end{cases} \quad (2.75)$$

where  $N_l$  is the number of operating conditions considered for the designed product. It is hypothesized that in all operating condition the structural boundary conditions will not change and the only load applied to the structure will change. Under this hypothesis the FEA

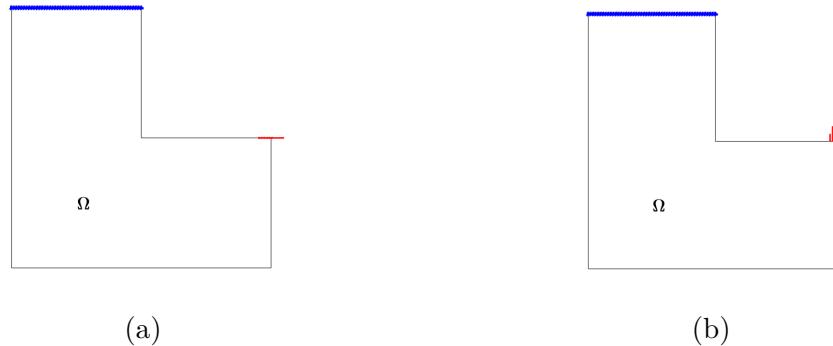


Figure 2.25: Multiple load cases applied to the L-shape, a) horizontal distributed load, b) vertical distributed load

model will only have several right hand side vectors, each one corresponding to a physical loading situation. To enhance the numerical efficiency it is possible to consider again a

smooth approximation of maximum function (a similar approach was proposed in [Le et al., 2010]) applied to all load cases that reduce the number of constraints to be considered in the optimization formulation:

$$\begin{cases} \min_{\{0\} \leq \{x\} \leq \{1\}} \{|\Omega_{el}|\}^T \{x\} \\ s.t. \\ \frac{1}{P_l} \log \left( \frac{\sum_{l_c=1}^{N_l} e^{P_l(G_{KS}^l)_{l_c}}}{N_l} \right) \leq 0 \end{cases} \quad (2.76)$$

One can observe that with this formulation one includes a double error approximating the

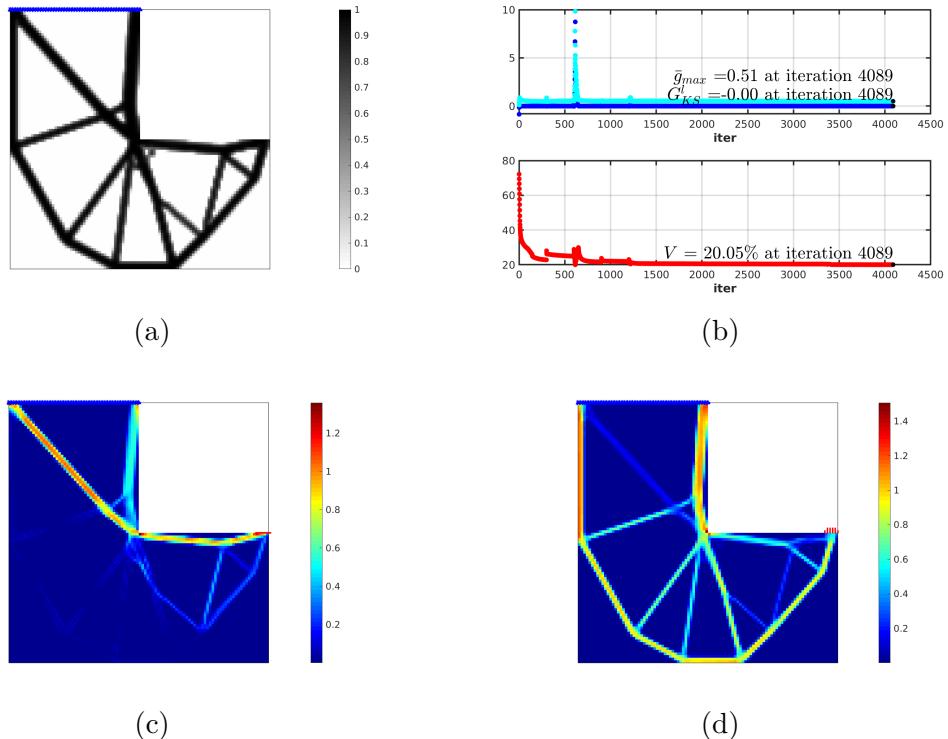


Figure 2.26: Results for the multiple load case study of an L-shape for a  $100 \times 100$  mesh. a) density plot, b) convergence plot, c)-d) von Mises stress plot for the first and the second load case respectively.

maximum, therefore the choice of  $P_l$  and  $P$  should be a good compromise between response non-linearity and maximum stress approximation accuracy. To show an application of this approach an L-shape design problem under 2 load cases was implemented c.f. figure 2.25. This study considers  $P = 10$  and  $P_l = 4$ . And the same settings of table 2.2. The study was conducted for a  $100 \times 100$  mesh and the results are reported in figure 2.26. The continuation strategy successfully enforces a black and white design in the solution c.f. figure 2.26a. On the other hand the convergence history was quite longer than the one experienced for the single load case stress based topology optimization and this is probably due to the additional non-linearity induced by load case aggregation. The nested use of KS aggregation reduces

the number of design constraint responses but cumulate the effect of nonlinearity of both stress and load case aggregation. In figures 2.26c and 2.26d one can observe von Mises stress distribution of the solution under each load case. It can be observed that the first load case adds some load paths that activate only in the first load case alleviating the maximal stress. The second load case is the most constraining with the final constraint violation of 0.51 in the second load case and of 0.3 in the first one. This is due to the consecutive KS aggregations that sum their approximation error. A continuation strategy to deal with such scenarios is proposed in section 3.9.

## 2.4 SIMP application to the pylon and engine mount design

A civil aircraft Powerplant System (PPS) primary structure has as a primary function to attach the engine to the aircraft wing. Furthermore the design of all the components of the PPS, ( pylon, engine mounts and nacelle) has also significant importance on the final integrated engine performance. For example the variation of the radial clearances at the blade tip of each stage (tip-clearances) affects engine time on wing, compressor surge margin and the thrust specific fuel consumption [Lattime and Steinetz, 2002, Benito et al., 2008]. Controlling the tip clearance variation due to engine maneuvers is for these reasons a major criterion considered during the PPS design. Including engine performance in the pylon design loop was investigated in Bettebghor et al. [Bettebghor et al., 2013]. In that study both engine casing and pylon sizing optimization were simultaneously tackled in order to find a better mass distribution between engine and PPS structures, while achieving a feasible design. On the other hand the engine mounts were considered in a fixed position so that only the engine casing thicknesses influenced directly tip clearances. In this work the optimization of the PPS structure under a fixed engine architecture is considered. The engine pylon topology optimization was also treated in [Remouchamps et al., 2011] considering both structural compliance under several load cases and aerodynamic drag. In Xue et al. [Xue et al., 2012] an Ant Colony Algorithm is deployed to optimize the front pylon mounts on the base of average stress under multiple load cases. This work is not focused on pylon aerodynamic performance, as it was done in [Remouchamps et al., 2011], since the design space shape is fixed. On the other hand the pylon to engine interface was included inside the design zone. This gives the solution more freedom that is also necessary to have an impact on both engine deformations and PPS primary structure mass. The design zone is much larger than the one considered in [Xue et al., 2012] and Ant Colony Algorithm was not implemented to tackle the optimization problem. The optimization formulation adopted for this work is a mass minimization with both stress and engine performance constraints. The fact of including these constraints is beneficial for the total lead time in the design process. In this way, the solution provided by the topology optimization phase will need minor modifications in order to achieve a feasible design that should satisfy both stress and performance requirements. This comes however at increased computational cost as the gradients of these two new constraints have to be computed.

The engine and the design zone (c.f. figure 1.4) are connected using an elimination approach that was developed in subsection 1.3.5.1. The final stiffness matrix of the entire model can

then be directly assembled as:

$$[K(\{x\})] = [K_{DZ}(\{x\})] + [K_E] \quad (2.77)$$

From here on,  $[K(\{x\})]$  will be referred to as the stiffness matrix after the application of boundary conditions and to  $[K_E]$  as the contribution of the engine stiffness to  $[K(\{x\})]$ . In the same way one can make the assembly of the load vector matrix as:

$$[F(\{x\})] = [F_{DZ}(\{x\})] + [F_E] \quad (2.78)$$

Where  $[F_{DZ}(\{x\})]$  is the matrix whose columns are load vectors coming from the design zone that can depend on the configuration<sup>10</sup> and  $[F_E]$  is the matrix whose columns are load vectors applied on the engine model. The static balance equations will be written as:

$$[K(\{x\})][U(\{x\})] = [F(\{x\})] \quad (2.79)$$

The so computed displacement matrix  $[U(\{x\})]$  is employed for the evaluation of design zone von Mises stress and the TSFC variation (c.f. subsection 1.2.2). The mass of the PPS structure is also an important parameter that was consider through the volume fraction defined as:

$$V(\{x\}) = \frac{\sum_{i=1}^{N_{el}} x_i |\Omega_i|}{\sum_{i=1}^{N_{el}} |\Omega_i|} = \frac{\{|\Omega|\}^T \{x\}}{\{|\Omega|\}^T \{\mathbf{1}\}} \quad (2.80)$$

Where  $|\Omega_i|$  is the Volume of the  $i^{th}$  element,  $\{|\Omega|\}$  is the vector containing the volume of each element and  $\{\mathbf{1}\}$  is the vector having the same length of  $\{x\}$  with 1 for each row. The value of  $V(\{x\})$  is between 0 and 1 and gives the fraction of volume that is filled by active material. The final design should be as light as possible and should reduce the engine consumption variation induced by aircraft maneuvers. To get reasonable results the von Mises stress in the design zone should also be lower than an allowable value. The full problem can thus be written as a non-linear constrained optimization problem (c.f. Eq. 2.81 for one load case), seeking to minimize the mass of the PPS structure, while imposing constraints on TSFC variations and maximum von Mises stress.

$$\left\{ \begin{array}{l} \min_{\{x\}} V(\{x\}) \\ \text{s.t.} \\ 0 \leq x_i \leq 1 \quad \forall i = 1, 2, \dots, N_{el} \\ G_T(\{U(\{x\})\}) = \frac{\Delta TSFC\%(\{U(\{x\})\}) - T_0}{T_0} \times 100 \leq 0 \\ [K(\{x\})]\{U(\{x\})\} = \{F(\{x\})\} \\ (\sigma_{VM})_j(\{x\}, \{U(\{x\})\}) \leq \sigma_{lim} \quad \forall j | x_{i(j)} > 0 \end{array} \right. \quad (2.81)$$

Where the  $(\sigma_{VM})_j$  is the von Mises stress computed in the  $j$ -th Gauss point and  $x_{i(j)}$  is the pseudo density of the  $i$ -th element that contains the  $j$ -th quadrature point, and  $\sigma_{lim}$  represents the allowable von Mises stress in the design zone and  $T_0$  is an allowable overconsumption due to the maneuver load case. Classical compliance minimization formulation is cheaper

---

<sup>10</sup>This is the case of acceleration induced load that depends on the mass of the solution and so on the configuration given by the design vector  $\{x\}$ .

than the proposed formulation, and can be adopted to get the inspiration for novel designs. Nevertheless the stiffest design will not always be able to respect both stress and engine consumption specifications. For this reason the formulation of equation 2.81, that also includes engine performance and stress constraints will be considered for the rest of this study. To evaluate the microscopic stress tensor in each Gauss point of the design zone, in the stiffness assembly phase the product of stress deformation matrix  $[D]|_{x=1}$  and of the displacement-deformation matrix  $[B]$  has to be saved in  $[DB]$ . This large sparse matrix reads in terms of input a displacement vector and provides as an output a stress vector containing 6 stress tensor components for each Gauss point of the design zone:

$$\{\bar{\sigma}(\{U(\{x\})\})\} = [DB]\{U(\{x\})\} \quad (2.82)$$

The von Mises stress in 3D is then:

$$\sigma_{VM} = \sqrt{\sigma_{xx}(\sigma_{xx} - \sigma_{yy}) + \sigma_{yy}(\sigma_{yy} - \sigma_{zz}) + \sigma_{zz}(\sigma_{zz} - \sigma_{xx}) + 3(\tau_{xy}^2 + \tau_{yz}^2 + \tau_{zx}^2)} \quad (2.83)$$

Using the same step adopted for the 2D use case (c.f. equations (2.58), (2.60)) , the satisfaction of stress constraints is thus imposed in problem (2.81) as:

$$\left\{ \begin{array}{l} \min_{\{x\}} V(\{x\}) \\ \text{s.t.} \\ G_T(\{U(\{x\})\}) = \frac{\Delta TSFC\%(\{U(\{x\})\}) - T_0}{T_0} \times 100 \leq 0 \quad \forall i = 1, 2, \dots, N_{el} \\ [K(\{x\})]\{U(\{x\})\} = \{F(\{x\})\} \\ G_{KS}^l(\{x\}, \{U(\{x\})\}) \leq 0 \end{array} \right. \quad (2.84)$$

### 2.4.1 Sensitivity computation

This subsection gives more details about the computation of responses sensitivities in problem (2.84). The reader can observe that a major complexity with respect to the 2D use-case comes from the fact that the mesh is non uniform. It is straightforward to use equations (2.34) and (2.35) to compute  $\left\{ \frac{dG_T}{dx} \right\}$ ,  $\left\{ \frac{dV}{dx} \right\}$  and  $\left\{ \frac{dG_{KS}^l}{dx} \right\}$ . Let's start by the evaluation of the TSFC variation constraint  $G_T$ . Since  $\Delta TSFC\%$  has no direct dependency on the  $\{x_{Phys}\}$  it is possible to write:

$$\left\{ \frac{\partial G_T}{\partial x_{Phys}} \right\} = \{0\} \quad (2.85)$$

By the use of equations 1.43,1.44 and 1.45:

$$\left\{ \frac{\partial G_T}{\partial U} \right\} = \frac{100}{T_0} \left\{ \frac{\partial \Delta TSFC\%}{\partial U} \right\} = \frac{100}{T_0} \sum_{s=1}^{ns} \lambda_{(s)} \left\{ \frac{\partial R_{(s)}}{\partial U} \right\} = \frac{100}{T_0} \sum_{s=1}^{ns} \frac{\lambda_{(s)}}{N_{(s)} R_{(s)}} [\gamma_{(s)}]^T \delta_{(s)} \quad (2.86)$$

Similarly, since  $V$  is only dependent on  $x_{Phys}$ :

$$\left\{ \frac{\partial V}{\partial U} \right\} = \{0\} \quad (2.87)$$

By the use of equation (2.80):

$$\left\{ \frac{\partial V}{\partial x_{Phys}} \right\} = \frac{\{|\Omega|\}}{\{|\Omega|\}^T \{\mathbf{1}\}} \quad (2.88)$$

The lower bound Kreisselmeier-Steinhauser function both depends on  $\{x_{Phys}\}$  and  $\{U(\{x_{Phys}\})\}$  so that:

$$\left\{ \frac{\partial G_{KS}^l}{\partial x_{Phys}} \right\} = \frac{\sum_{i=1}^{N_G} \left\{ \frac{\partial \bar{g}_i}{\partial x_{Phys}} \right\} e^{P\bar{g}_i}}{\sum_{i=1}^{N_G} e^{P\bar{g}_i}} \quad (2.89)$$

With  $\left\{ \frac{\partial \bar{g}_i}{\partial x_{Phys}} \right\}$  defined as:

$$\left\{ \frac{\partial \bar{g}_i}{\partial x_{Phys}} \right\}_j = \begin{cases} \frac{(\sigma_{VM})_i}{\sigma_{lim}} - 1 & i \in \mathbf{G}_j \\ 0 & i \notin \mathbf{G}_j \end{cases} \quad (2.90)$$

$\mathbf{G}_j$  referred to the  $j$ -th element Gauss point index. In the same way one can evaluate:

$$\left\{ \frac{\partial G_{KS}^l}{\partial U} \right\} = \frac{\sum_{i=1}^{N_G} \left\{ \frac{\partial \bar{g}_i}{\partial U} \right\} e^{P\bar{g}_i}}{\sum_{i=1}^{N_G} e^{P\bar{g}_i}} = \frac{\sum_{i=1}^{N_G} \left\{ \frac{\partial(\sigma_{VM})_i}{\partial U} \right\} \frac{(x_{Phys})_i}{\sigma_{lim}} e^{P\bar{g}_i}}{\sum_{i=1}^{N_G} e^{P\bar{g}_i}} \quad (2.91)$$

From von Mises stress definition (2.83) in each Gauss point  $i$  it is possible to write:<sup>11</sup>

$$\begin{aligned} \left\{ \frac{\partial(\sigma_{VM})}{\partial U} \right\} = & \frac{1}{2(\sigma_{VM})} \left( (2\sigma_{xx} - \sigma_{yy} - \sigma_{zz}) \left\{ \frac{\partial \sigma_{xx}}{\partial U} \right\} + \right. \\ & + (2\sigma_{yy} - \sigma_{zz} - \sigma_{xx}) \left\{ \frac{\partial \sigma_{yy}}{\partial U} \right\} + \\ & + (2\sigma_{zz} - \sigma_{xx} - \sigma_{yy}) \left\{ \frac{\partial \sigma_{zz}}{\partial U} \right\} + \\ & \left. + 6 \left( \tau_{xy} \left\{ \frac{\partial \tau_{xy}}{\partial U} \right\} + \tau_{xz} \left\{ \frac{\partial \tau_{xz}}{\partial U} \right\} + \tau_{yz} \left\{ \frac{\partial \tau_{yz}}{\partial U} \right\} \right) \right) \end{aligned} \quad (2.92)$$

In this equation each vector,  $\left\{ \frac{\partial \sigma_{xx}}{\partial U} \right\}, \left\{ \frac{\partial \sigma_{yy}}{\partial U} \right\}, \left\{ \frac{\partial \sigma_{zz}}{\partial U} \right\}, \left\{ \frac{\partial \tau_{xy}}{\partial U} \right\}, \left\{ \frac{\partial \tau_{xz}}{\partial U} \right\}, \left\{ \frac{\partial \tau_{yz}}{\partial U} \right\}$  corresponds for equation (2.82) to a column of  $[DB]^T$ . For this reason equation (2.91) can be written as:

$$\left\{ \frac{\partial G_{KS}^l}{\partial U} \right\} = \frac{1}{\sum_{i=1}^{N_G} e^{P\bar{g}_i}} [DB]^T \{\tilde{S}\} \quad (2.93)$$

---

<sup>11</sup>Here the Gauss point index  $i$  is not indicated for conciseness.

Where :

$$\{\tilde{S}\} = \begin{Bmatrix} \{\tilde{S}\}_1 \\ \{\tilde{S}\}_2 \\ \vdots \\ \{\tilde{S}\}_{N_G} \end{Bmatrix}, \quad \{\tilde{S}\}_i = \frac{(x_{Phys})_i e^{P\bar{g}_i}}{2\sigma_{lim}(\sigma_{VM})_i} \begin{Bmatrix} 2(\sigma_{xx})_i - (\sigma_{yy})_i - (\sigma_{zz})_i \\ 2(\sigma_{yy})_i - (\sigma_{zz})_i - (\sigma_{xx})_i \\ 2(\sigma_{zz})_i - (\sigma_{xx})_i - (\sigma_{yy})_i \\ 6(\tau_{xy})_i \\ 6(\tau_{xz})_i \\ 6(\tau_{yz})_i \end{Bmatrix} \quad (2.94)$$

The results of the sensitivity computations are summarized in table 2.3.

Table 2.3: Summary table of sensitivities terms needed for equations (1.73) and (1.74) to compute  $\left\{ \frac{dO}{dx_{Phys}} \right\}$

$O$	$\left\{ \frac{\partial O}{\partial U} \right\}$	$\left\{ \frac{\partial O}{\partial x_{Phys}} \right\}$
$V$	$\{0\}$	$\frac{\{\Omega\}}{\ \Omega\ } \{1\}$
$G_T$	$\frac{100}{T_0} \sum_{s=1}^{ns} \frac{\lambda_{(s)}}{N_{(s)} R_{(s)}} [\gamma_{(s)}]^T \delta_{(s)}$	$\{0\}$
$G_{KS}^l$	$\frac{1}{\sum_{i=1}^{N_G} e^{P\bar{g}_i}} [DB]^T \{\tilde{S}\}$	$\frac{\sum_{i=1}^{N_G} \left\{ \frac{\partial \bar{g}_i}{\partial x_{Phys}} \right\} e^{P\bar{g}_i}}{\sum_{i=1}^{N_G} e^{P\bar{g}_i}}$

By the use of equations (2.40),(2.41) and (2.42):

$$\frac{d[K]}{d(x_{Phys})_i} = \frac{d[K_{DZ}]}{d(x_{Phys})_i} = \bigoplus_{el=i} p(E_{max} - E_{min}) (x_{Phys})_i^{p-1} \left[ K_{el}^{(1)} \right] \quad (2.95)$$

The present work only considers constant load, so that  $\left[ \frac{dF}{dx_{Phys}} \right] = 0$ .<sup>12</sup> For a general uniform acceleration (e.g. inertial loads) the sensitivities of the load vector could be computed using:

$$\{F_{DZ}\}_i = \int \rho f_i \Phi_i dV = \frac{1}{8} \sum_{j=1}^{m_i} (x_{Phys})_j f_i \Omega_j \rho \quad (2.96)$$

So that:

$$\{F_{DZ}\} = \frac{1}{8} [\rho f \Omega] \{x_{Phys}\} \quad (2.97)$$

Then:

$$\left[ \frac{dF}{dx_{Phys}} \right] = \left[ \frac{dF_{DZ}}{dx_{Phys}} \right] = \frac{1}{8} [\rho f \Omega] \quad (2.98)$$

The adjoint evaluation of sensitivities presented in this subsection, needs the use of equation (2.31) twice, once for the evaluation of  $\left\{ \frac{dG_T}{dx_{Phys}} \right\}$  and once for  $\left\{ \frac{dG_{KS}^l}{dx_{Phys}} \right\}$ . Therefore the stiffness matrix has to be inverted for 3 different right hand side vectors per each optimization loop iteration. Taking the formulation described here in matrix notation also facilitates implementation in the presented Matlab framework, taking advantage from vectorization.

<sup>12</sup>This hypothesis could be justified by the fact that the inertial loads for the pylon itself is only a small part of the total load applied to the pylon structure.

### 2.4.2 Single load case analysis

The optimization framework developed was applied on a problem involving the engine model described in section 1.2.1 and on the design mesh represented in figure 2.27a and for the axial load case showed in figure 2.30. In table 2.4 one can find the optimization set-up details.<sup>13</sup>

Table 2.4: Optimization problem set up. The hypothesis made to get numerical results are listed here

Symbol	Name	Value
$E_0$	Young's Modulus	210 GPa
$\nu$	Poisson Ratio	0.29
$N_{el}$	Number of elements in the design zone	486400
$N_{DOFs}$	Number of rows of the stiffness matrix	1529847
$N_G$	Number of Gauss points in the design zone	3891200
$r$	filtering radius	2 × mesh average size
$p$	SIMP penalty	3
$P$	Aggregation constant	4
$\sigma_{lim}$	Allowable stress to be used in eq. (2.58)	10 MPa
$\sigma_{alw}$	Maximum local allowable stress	47.9 MPa
$T_0$	Allowable consumption variation	0.15 %
$\{\rho f\}$	Inertial load	{0}
	Stopping condition on the Karush-Kuhn Tucker residual norm	$KKTn \leq 10^{-3}$
	Stopping condition on the iteration number	$iter \geq 300$
	MMA external move limit for first 2 iterations	0.4
	MMA maximum asymptote distance from the current point	0.1

The mesh refinement procedure introduced in 1.4.2.3 was applied twice to the original mesh imported from Abaqus c.f. Fig. 2.16a. The final stiffness matrix has 1.5 Million DOFs before applying boundary conditions. The filtering radius was taken as 2 times the average element size that is corresponding to twice the average size of the original mesh . The SIMP penalty value was set to 3 and the stress constraint aggregation constant to 4. This is a relatively small value for P that improves optimization convergence which requires the use of a scaling factor on the stress allowable. In fact from equation B.3, it is possible to conclude that  $\bar{g}_{max} - G_{KS}^l < \frac{\ln(N_G)}{P} = \frac{\ln(3891200)}{4} \approx 3.7936$ . So that even if the  $G_{KS}^l \leq 0$  this will only imply that  $\bar{g}_{max} < 3.7936$ . Assuming that the maximum relaxed stress constraint violation is on a material with  $x_i = 1$  this means that the actual maximum von Mises stress allowable is  $\sigma_{max} < (1 + 3.7936) \times \sigma_{lim} = \sigma_{alw} = 47.9MPa$ , that is the value of stress that has to be avoided even locally. In this example  $\sigma_{lim} = \frac{\sigma_{alw}}{1+3.7936} \approx 10MPa$  that is corresponding to a maximum local stress not to be exceeded of  $\sigma_{alw} = 47.9MPa$ . The initial design consists of  $x_i = 1, \forall i = 1, 2, \dots, N_{el}$ , the allowed TSFC variation was 0.15%, that is a 12% improvement from the initial design. Stress constraint nonlinearity can be

<sup>13</sup>MMA maximum asymptote distance from the current point value has not a particular name in the mmasub Matlab function provided by Svanberg [Svanberg, 2004]. It can be found in the lines where lowmin and uppmax variables are computed as the coefficient that multiplies the variable range. Reducing this value the algorithm behaves more conservatively when approximating the real functions overestimating their convexity.

source of MMA convergence difficulties and sometimes divergence. To tackle this problem here is proposed to set a smaller value of the MMA external move limit (here considered as the maximum difference between the asymptotes distance from the configuration point). This imposes MMA to produce conservative local approximations of the original optimization problem that are less prone to violate optimization constraints. The convergence history of

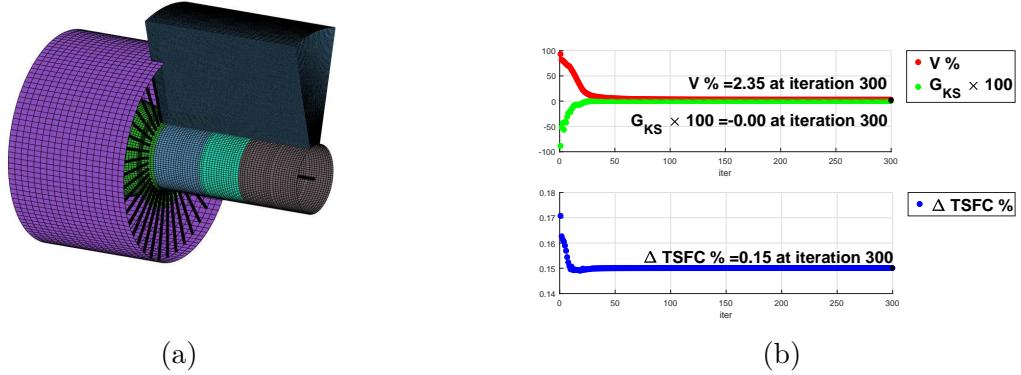


Figure 2.27: (a) Design zone mesh used for topology optimization problem. Each element of the original element is cut in 64 new elements, which leads to a total of  $7600 \times 64 = 486400$  8-node linear 3D solid finite elements and 509949 nodes. (b) Convergence history of  $\Delta TSFC\%$ ,  $V\%$ ,  $G_{KS}^l \times 100$  after 300 design iterations

volume fraction,  $\Delta TSFC\%$  and of  $G_{KS}^l$  is presented in figure 2.27b and the final design configuration is presented in figure 2.27a. Note that just considering figure 2.27b one could conclude that convergence was achieved approximately after 50 iterations. This is because the stress constraints are very non-linear so that MMA needs to keep the optimization step very small in order to avoid stress constraint violation. Stopping the optimization after 50 iteration would lead to a non-converged design full of gray elements. In the same way considering the design variable variation as stopping criterion could lead again to gray solutions. The optimization was therefore stopped after 300 design iterations. A KKT norm condition of 0.001 was also considered but was not achieved in the maximum number of iterations. The final design is well connected and respects constraints. It is possible to find two main load paths, one at the front of the engine and a second at the rear. Moreover engine casing reinforcement structures can be found at the front of the solution to avoid tip clearance variations. To make displacement and stress plots in figure 2.29a and 2.29b the solution was thresholded i.e.:

$$x\bar{P}hys_i = \begin{cases} 1 & \text{if } xPhys_i \geq t_{sh} \\ 0 & \text{otherwise.} \end{cases} \quad (2.99)$$

Where  $xPhys$  are the physical densities. With  $t_{sh} = 0.22$ , selected in order to form a well-connected solution. Doing so the final solution performance are deteriorated as summarized in table 2.5:

The thresholded solution still respects von Mises stress constraint, on the other hand the allowable fuel consumption constraint is violated by 1.47% and the final volume fraction is increased by 34%. Even after this increase, the final volume fraction can still be considered

Table 2.5: Solution responses before and after thresholding

Response	Original solution	After thresholding ( $t_{sh} = 0.22$ )
$\Delta TSFC\%$	0.15	0.1522
$V\%$	2.35	3.15
$G_{KS}^l$	$-4.3 \times 10^{-6}$	$-9 \times 10^{-3}$
$\max(\sigma_{VM})$	25.37 MPa	22.98 MPa

satisfactory for the sake of this study.

von Mises stress color maps, based on the average over Gauss points are presented in figure 2.29a. As expected the final design has a final maximum stress that is greater than the 10 MPa imposed through the  $G_{KS}^l$  function, but lower than 47.9 MPa as a consequence of the choice of  $P$ . The solution displacement field 2.29b is consistent with boundary conditions and to the load applied to the structure. The final von Mises stress is obviously not homogeneous within the solution. It reaches its maximum in the regions adjacent to the wing and at the interface with the engine model. Note that the high stresses at the attachment with the wing are induced by the geometry and loading. On the other hand the high stresses at the interface with the engine model are numerical artefacts of the kinematic tying approach of the design zone and the engine superelements. In fact RL-RBF were employed in this example. More complex kinematic tying approaches such as the ones reviewed in chapter 1 could be implemented to alleviate this issue. The structure found by the optimization algorithm is nearly planar and contained in the x-z plane. This is due to the load considered here (axial load) and to the symmetry of the engine model. This solution can therefore be further improved considering multiple load cases that will load the structure in different directions. Changing mesh tying approaches used to deal with the mesh inconsistency at the interface with the engine could also potentially improve the solution. The results found are not of direct practical interest due to several simplifying assumptions considered in the models being used (especially the simplified engine model). Moreover due to the formulation adopted, the solution can use the stiffness of the engine casing, possibly compromising the engine sizing. Nevertheless the solution is consistent with the model hypothesis and shows that it is possible to deal with both engine deformation and stress criteria in the same 3D topology optimization framework.

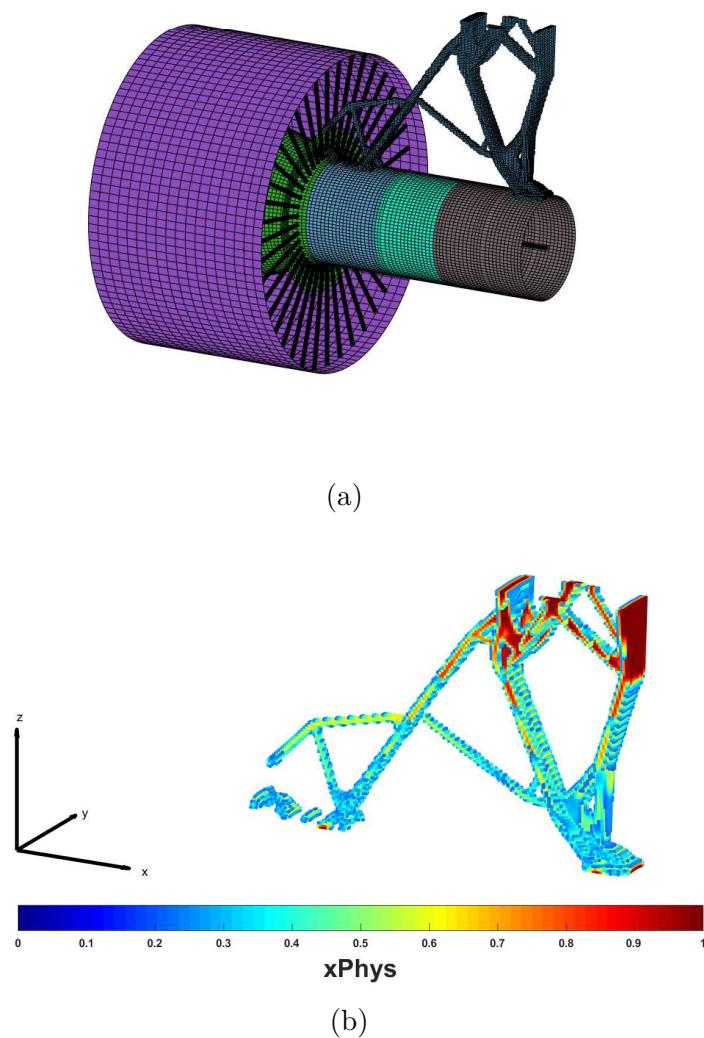


Figure 2.28: Topology optimization results. (a) Design configuration after 300 iterations, only densities greater than 0.22 are displayed, (b) Physical density color map

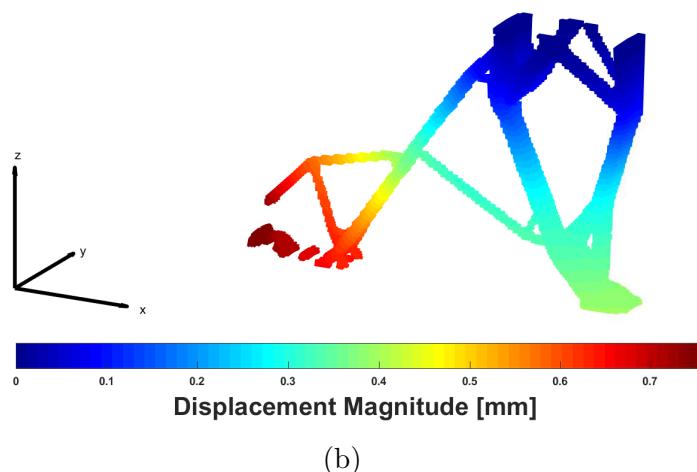
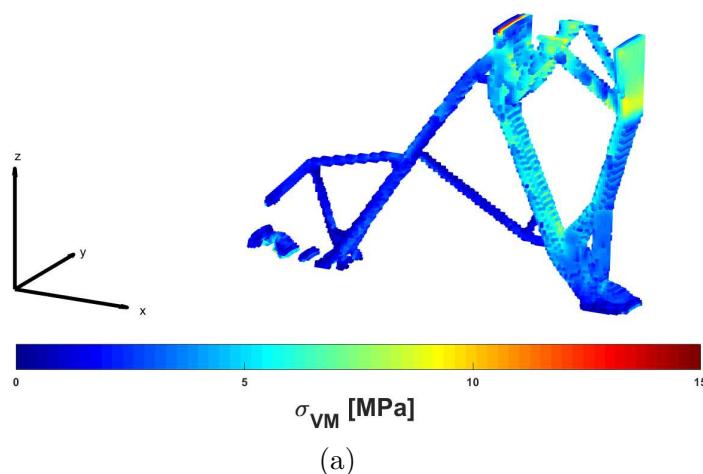


Figure 2.29: Topology optimization results. (a) von Mises Stress color map, (b) Displacement magnitude color map.

### 2.4.3 Multiple load case analysis with symmetric solutions

To achieve improved design here the previous analysis is extended to multiple loads applied to the engine model (c.f. figure 2.30). Moreover, solutions will be enforced to be symmetric with

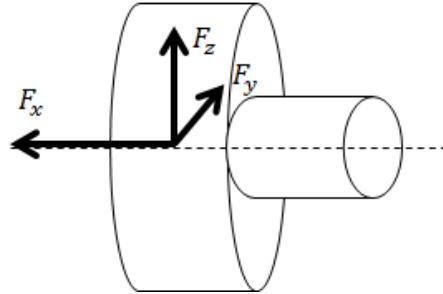


Figure 2.30: Load considered for the Pylon and engine mounts topology optimization. The  $F_y$  and  $F_z$  forces are distributed on the Fan casing.  $F_x$  is the same load case considered for the single load case analysis.

respect to the  $xz$  plane.<sup>14</sup> To deal with the multiple loads considered a KS approximation of both maximum of stress constraint violation and maximum of TSFC variation under each load case:

$$\begin{cases} \min_{\{x\}} V(\{x\}) \\ \text{s.t.} \\ 0 \leq x_i \leq 1 & \forall i = 1, 2, \dots, N_{el} \\ G_T = \frac{\frac{1}{P_l} \log \left( \frac{\sum_{l_c=1}^{N_l} e^{P_l(\Delta TSFC\%) l_c}}{N_l} \right) - T_0}{T_0} \times 100 \leq 0 \\ [K(\{x\})][U] = [F(\{x\})] \\ \frac{1}{P_l} \log \left( \frac{\sum_{l_c=1}^{N_l} e^{P_l(G_{KS}^l) l_c}}{N_l} \right) \leq 0 \end{cases} \quad (2.100)$$

In order to get only symmetrical solutions one can introduce a simple mapping between the left ( $y_c \geq 0$ ) and the right side ( $y_c < 0$ ) of the model, where  $y_c$  is the coordinates of the element centroids. In this way one could write that:

$$\{x\} = \left\{ \begin{array}{l} \{x_l\} \\ \{x_r\} \end{array} \right\} = \left[ \begin{array}{c} [I] \\ [\mathbf{S}] \end{array} \right] \{x_l\} \quad (2.101)$$

Where  $[\mathbf{S}]$  is the symmetrical mapping whose elements are only 0 and 1 and  $\{x_l\}$  and  $\{x_r\}$  represent the left and the right design variable vectors, corresponding to the left and the

<sup>14</sup>This requirement is consistent with the fact that left and right wing pylons primary should be identical (to reduce the manufacturing costs) and that only half of the operating load conditions are applied to the structure to make its sizing.

right part of the design space respectively.<sup>15</sup> The optimization variables become the only  $\{x_l\}$  vector so that the number of design variables are reduced to 243200. For the load case aggregation a value of  $P_l = 10$  was selected. When aggregating over the load cases a supplementary error of  $\frac{\log(N_l)}{P_l}$  is added to the previously introduced for the Gauss point aggregation.<sup>16</sup> For this reason the value of the  $\sigma_{alw}$  was increased to 49 MPa. The initial guess is again a uniform distribution of full material. The allowable TSFC variation was this time 0.16%, that is still an improvement 11% with respect to the baseline design. In table 2.6 one can find the optimization set up details. The interested reader can find several solutions obtained for different values of the TSFC variation constraint in section B.3. Actually due to the aggregation over load cases by the KS function approximation, the final maximum variation of consumption is equivalent to the one of the first design (0.17%).

Table 2.6: Optimization problem set up. The hypothesis made to get numerical results on the multiple load cases are listed here

Symbol	Name	Value
$E_0$	Young's Modulus	210 GPa
$\nu$	Poisson Ratio	0.29
$N_{el}$	Number of elements in the design zone	486400
$N_{DOFs}$	Number of rows of the stiffness matrix	1529847
$N_G$	Number of Gauss points in the design zone	3891200
$N_{var}$	Number of Design variables	243200
$r$	filtering radius	2 × mesh average size
$p$	SIMP penalty	3
$P$	Aggregation constant	4
$P_l$	Aggregation constant for load case	10
$\sigma_{lim}$	Allowable stress to be used in eq. (2.58)	10 MPa
$\sigma_{alw}$	Maximum local allowable stress	49 MPa
$T_0$	Allowable consumption variation	0.16 %
$\{\rho f\}$	Inertial load	{0}
	Stopping condition on the Karush-Kuhn Tucker residual norm	$KKTn \leq 10^{-3}$
	Stopping condition on the iteration number	$iter \geq 300$
	MMA external move limit for first 2 iterations	0.4
	MMA maximum asymptote distance from the current point	0.1

The convergence history for  $\Delta TSFC\%$ ,  $G_{KS}^l$  and  $V$  is shown in figure 2.31, The iso-contour plot of the solution is shown in figure 2.32, the density plot can be found in figure 2.33 and Displacement and von Mises stress plot for each load case are shown in figure 2.34. The final design is well connected. Moreover the load paths can be identified in figure 2.34b,

<sup>15</sup>The reader can observe that such procedure can only be applied to symmetric design zone meshes. Extension to non symmetric meshes would require a further interpolation of one half mesh with the other.

<sup>16</sup>In appendix B.2 it is shown that for  $N_G$  entries  $\bar{g}_{max} - G_{KS}^l < \frac{\ln N_G}{P}$ . In the situation of multiple load case it is necessary to approximate  $\bar{g}_{max} = \max_{l=1,2,\dots,N_l}(\max_{i=1,2,\dots,N_G}(\bar{g}_{il}))$ . In the proposed approach first the first maximum function is approximated by a KS function per load case. This introduce a first error lower than  $\frac{\ln N_G}{P}$  to each KS approximation of the maximum over the number of Gauss point. When approximating the maximum of the previously computed KS function over the load cases this approximation introduce a second error this time lower than  $\frac{\ln N_l}{P_l}$ .

2.34d, 2.34f under each load case. It is possible to recognize two main structures, one that connects the engine core part with the frontal part of the boundary conditions and a second structure that support the backbone bending effect under the thrust load. In the frontal part one can also recognize components typically found in pylon as two lateral panel whose main function is to bear the lateral load (in this case  $F_y$ ) and a lower spar and an upper spar. On the engine casing one can also recognize stiff structures that redistribute loads. That probably aims at reducing fuel consumption variation induced by casing out of roundness effects. Due to the geometry chosen for the boundary conditions and the design zone this is still far from being a realistic solution but the design can be considered as a reasonable outcome considering the model hypothesis. For the thresholded solution it was selected a value of  $t_{sh} = 0.45$ . Doing so the final solution performance are deteriorated as summarized in table 2.7.

The thresholded solution is slightly heavier than the original solution (+1.01%) but satisfies

Table 2.7: Solution responses before and after thresholding

Response	Original solution	After thresholding ( $t_{sh} = 0.45$ )
$\Delta TSFC\%$	0.16	0.1591
$V\%$	4.92	4.97
$G_{KS}^l$	$-6.3 \times 10^{-6}$	$-1.2 \times 10^{-2}$
$\max(\sigma_{VM})$	28.89 MPa	30.08 MPa

all the constraints. Stress displayed in figure 2.34 is again based on the average over Gauss points. Again artificial high stress can be found at the interface with the engine, and same discussion applies as previously. Even if this solution represents an improvement due to its symmetry and to its improved mechanical behavior with lateral load, in an industrial context such design, would still require further simplification in order to be considered as a feasible design. As the pylon needs to be assembled on the engine, an isostatic system of attachment should be preferred. Moreover curved spars and panels like the one in the design, represent a challenge for standard manufacturing procedures. To try to address such constraints, alternative topology optimization methods are described and investigated in chapter 3.

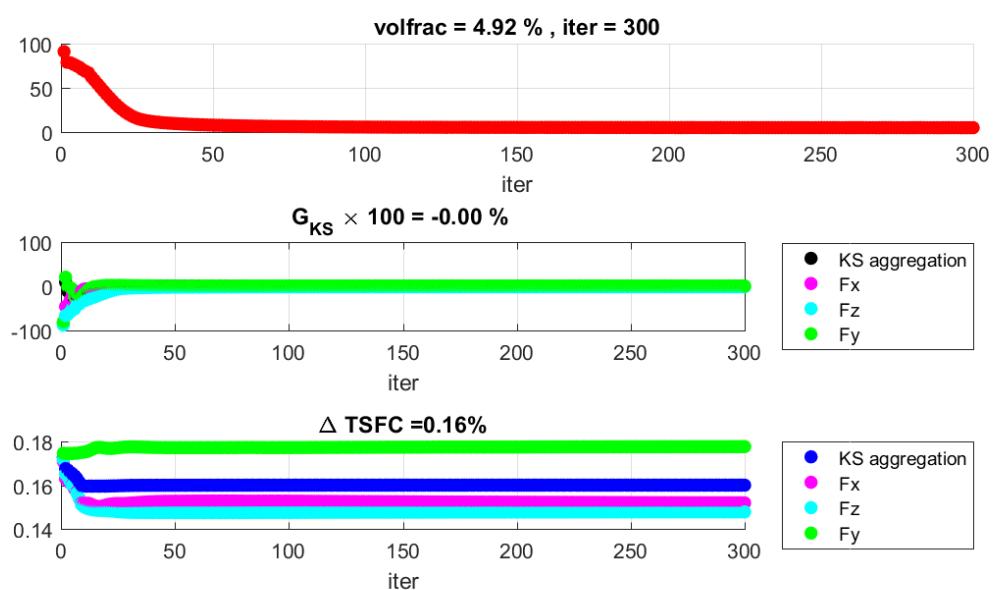


Figure 2.31: Convergence history of the multi load case topology optimization

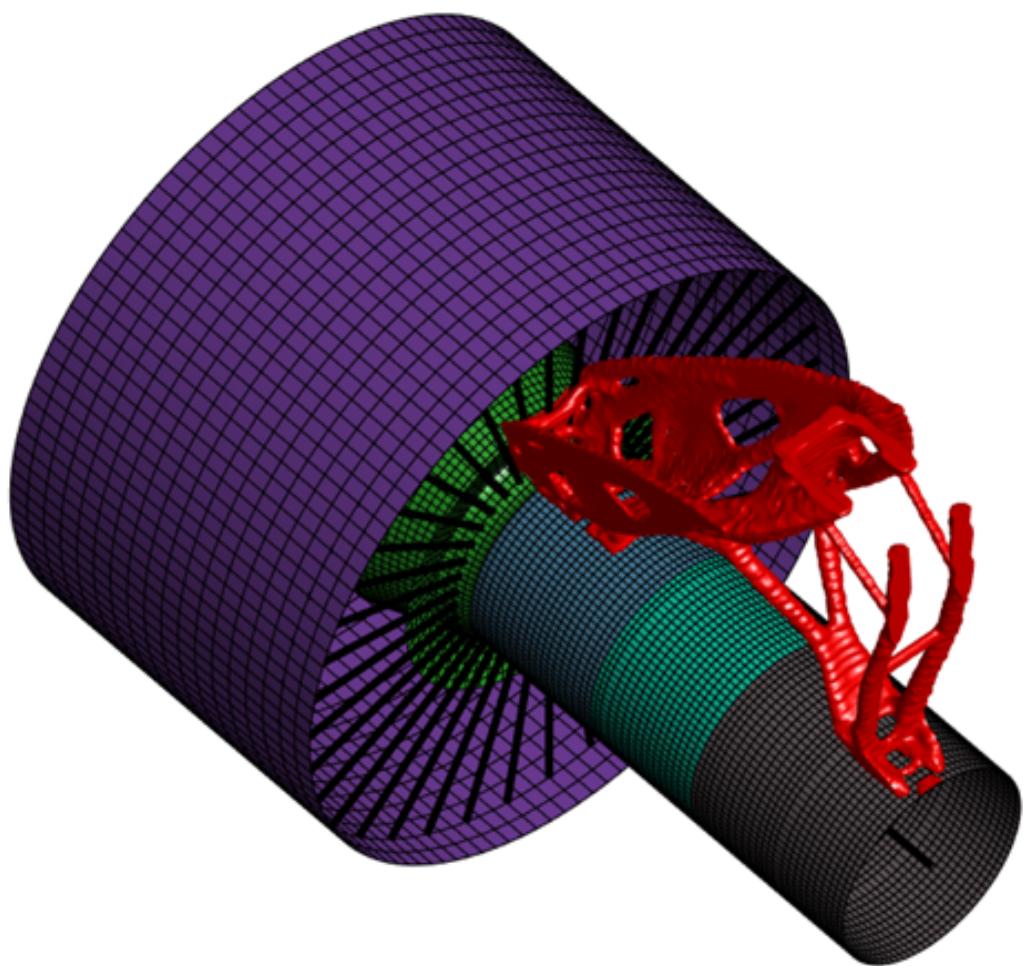


Figure 2.32: Iso-contour plot of the solution with a design threshold of 0.45

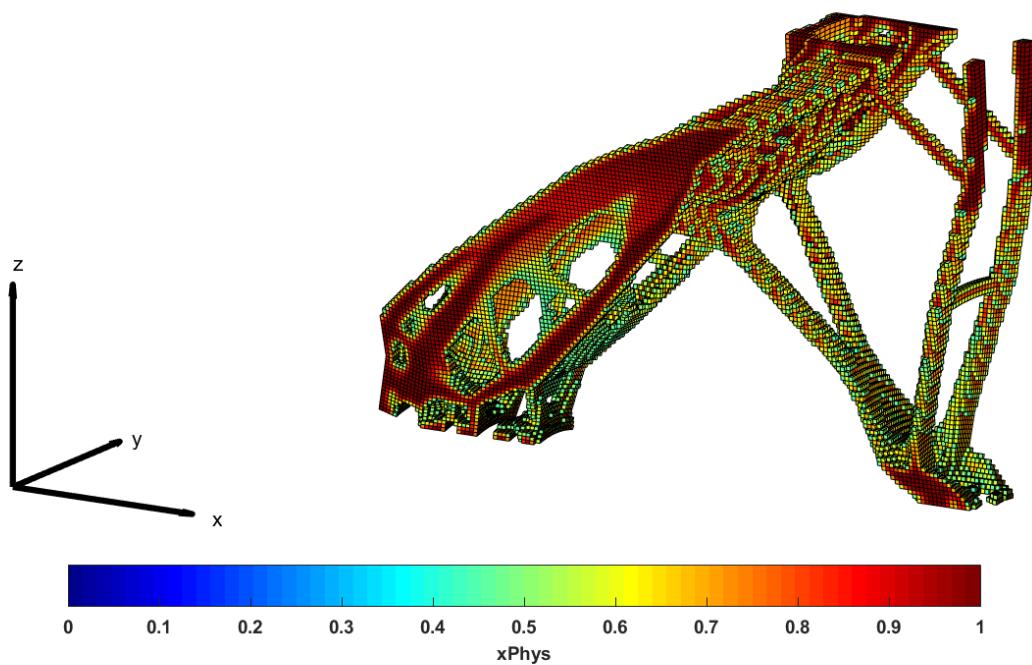


Figure 2.33: Density plot of the solution of the multi load case analysis for the pylon and engine mount design. Only elements with a physical density greater than 0.45 are represented.

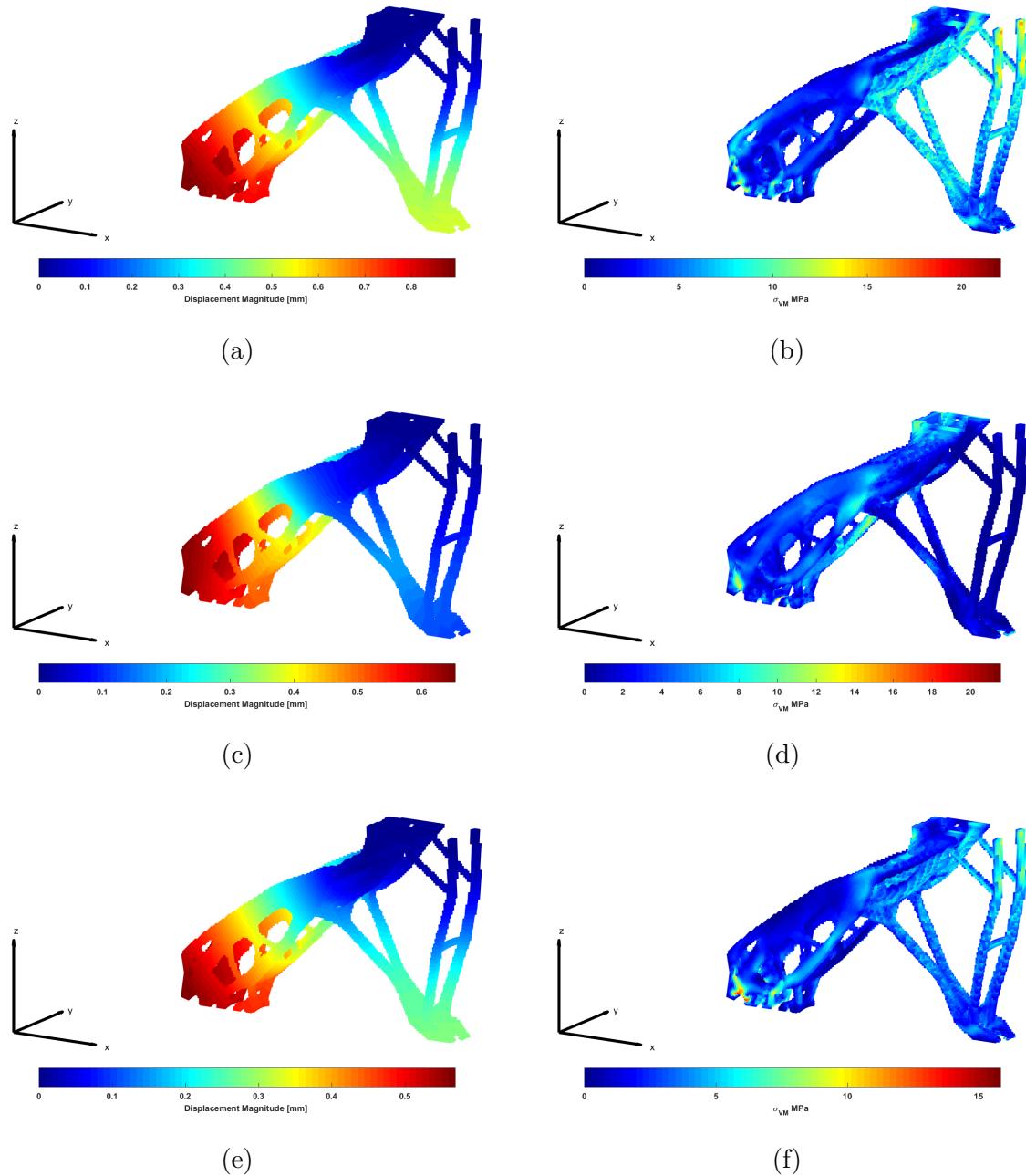


Figure 2.34: Topology optimization results. Displacement magnitude color map (a) for  $F_x$ , (c) for  $F_y$ , (e) for  $F_z$  respectively. von Mises stress color map (b) for  $F_x$ , (d) for  $F_y$ , (f) for  $F_z$  respectively.

## 2.5 Summary and conclusions

In this chapter the simulation driven design and density based topology optimization were reviewed. The optimization framework proposed in this thesis uses a simulation model recursively inside an optimization loop to achieve improved design with respect to fuel consumption variation and stress constraints. The Solid Isotropic Material with Penalization (SIMP) approach to topology optimization was reviewed to be included in the proposed framework. To solve this problem efficiently in the context of industrial applications, Sequential Convex Programming (SCP) algorithms and in particular the Method of Moving Asymptotes (MMA) were reviewed. Analytical sensitivities derivation was proposed by the adjoint approach to provide MMA solver with responses' derivatives. The unified aggregation relaxation approach was implemented to achieve singular optima and to deal with a large number of stress constraints in the topology optimization problem. A novel strategy for the evaluation of the filter matrix was proposed. This helps in reducing the computational burden required by filter matrix evaluation. Finally a formulation based on the mass minimization with constraints on the allowable von Mises stress and the fuel consumption variation was proposed to design the engine integration of a representative usecase. This Chapter contributes to the main goals of the PhD providing the theoretical bases for the development of Eulerian approaches to topology optimization and their efficient solution. To summarize the review covered the following topics:

- Structural optimization design loop
- Sequential convex programming
- Sensitivity computation by adjoint
- Solid Isotropic Material with Penalization method
- The density filter
- Stress based topology optimization by the Unified Aggregation Relaxation approach.

The reviewed strategies were implemented on a 2D use case problem in a framework developed from the 88 lines Matlab code of [Andreassen et al., 2011]. Results of stress based topology optimization problems are consistent with the findings of [Verbart et al., 2017] and were also extended to multiple load cases problems. The main contributions in this chapter consist in:

- Proposing a multigrid strategy for the evaluation of filter matrix for the sensitivity filter.
- The adaptation of this approach to multiple load cases
- The implementation of TSFC variation in the optimization formulation.
- The analytical derivation of derivatives using the adjoint method.

- This work led to the publication of a journal paper [Coniglio et al., 2019a] and two conference papers [Coniglio et al., 2017, Coniglio et al., 2018b].

Some challenges that further need to be addressed concern:

- The enforcement of manufacturing constraints. This is a very well documented problem in topology optimization and recently solutions for several kinds of technology are proposed in the literature [Liu et al., 2018b].
- The consideration of other failure modes in the optimization, as buckling and resonant frequencies.
- the consideration of stress constraints also in the engine model. Or equivalently control the load introduced on the engine structure.
- The reduction of solution interpretation time. Advanced tools for the post processing are available in industrial software as Inspire by Altair. Still, this requires human interventions and design simplifications, impacting the final solution's performance.
- The reduction of the total computational burden. The total CPU time is principally affected by the solution of state equations. This work addressed this problem by the use of iterative approaches and efficient preconditioners. The use of distributed memory architecture [Aage et al., 2017] and GPU accelerators [Wadbro and Berggren, 2009] would also benefit this framework.
- Solution dependence on the initial guess. This is quite an open problem in structural optimization of non-convex problems. The use of global approaches [Simon, 2013] is a possibility but the cost and the number of design variables are limits to their application to topology optimization problems.

Some of these challenges are partly addressed in the next chapter by the introduction of a different family of topology optimization approaches. The other research axes are to be studied in future works.



# Lagrangian Structural Topology optimization

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## Résumé

Dans ce chapitre nous passons en revue les approches d'optimisation topologique dites lagrangiennes ou bien explicites. Dans cette famille d'approches la solution est décrite par un ensemble de composantes de forme simple dont les dimensions, positions et orientations sont données comme variables de design. Entre les approches qu'on peut retrouver dans la littérature on considère:

- **MMC:** *Methode of Moving Morphable Components* avec modèle matériau d'ersatz.
- **GP:** *Geometry Projection*
- **MNA:** *Moving Node Approach*

Une nouvelle techniques appelée *Generalized Geometry Projection (GGP)* [Coniglio et al., 2019b] est proposée. Cet approche est en réalité une généralisation de l'approche de GP qui permet de représenter toutes les techniques mentionnées précédemment. En plus cette technique offre la possibilité d'éviter l'apparition de points d'inflexion dans les réponses obtenues utilisant la projection sur maillage éléments finis. Ensuite une étude bibliographique et comparative des approches les plus courantes pour l'assemblage géométrique de composantes est aussi proposée. Le calcul analytique des gradients est ensuite déduit. Différents cas simplifiés démontrent l'efficacité de l'approche proposée pour éviter les points d'inflexion. Ensuite les contraintes de stress sont aussi traitées par la formulation unifiée d'agrégation et relaxation de [Verbart et al., 2017]. Finalement l'application de l'approche MNA au problème de design du mât et attaches moteur, déjà traité dans le chapitre 2 par la méthode SIMP, est considérée. D'abord des composantes rectangulaires ensuite des composantes spécifiques au problème en étude sont traitées.

SIMP direct density based approaches to topology optimization [Bendsøe, 1989, Zhou and Rozvany, 1991, Bendsøe and Sigmund, 1995], such as the one in chapter 2, are currently among the most popular ones. The exploration power of these approaches, their freedom and their ease in handling large changes in the design configuration without re-meshing come however at the cost of some drawbacks as well:

- The number of design variables and of degrees of freedoms in the finite element model grows quite fast in 3D problems and their resolution becomes quickly prohibitive especially if considering optimization formulation that do not only include compliance and volume fraction.
- The typical "organic" designs obtained with direct density based methods, even if highly effective, may not be easily compatible with manufacturing requirements (e.g. casting, rolling, overhang angle constraints, etc.). Indeed as analyzed in the recent survey by Liu and Ma [Liu and Ma, 2016], and more recently of Liu et al. [Liu et al., 2018b] the implementation of manufacturing considerations in topology optimization is still a challenging, highly active area of research.
- Intermediate densities often subsist in the final design in direct density based approaches due to computational cost constraints, preventing the full convergence towards black and white designs. The gray elements would then have to be threshold to black and white solutions, which can lead to a loss in performance compared to the optimum.
- The density formulation of the topology optimization problem is ill posed and need to be transformed into a well posed one through restriction to feasible set of solutions

[Borrvall, 2001]. Some approaches like mesh independent filter techniques [Sigmund, 1994], depending on the filter size, can be computationally expensive.

An alternative family of approaches seeks to both use geometric primitives (i.e. geometric components) to define the optimal solution and inherit a fixed mesh as typical in direct density based topology optimization such as SIMP. In the literature one can find two major groups of approaches incorporating geometric features in topology optimization:

- A first group combines the free-form topology optimization with embedded components or holes shaped as geometric primitives. (c.f. Chen et al. [Chen et al., 2007]; Qian and Ananthasuresh [Qian and Ananthasuresh, 2004]; Xia et al. [Xia et al., 2012]; Zhang et al. [Zhang et al., 2011, Zhang et al., 2015]; Zhou and Wang [Zhou and Wang, 2013]; Zhu et al. [Zhu et al., 2008])
- A second group, to which this work belongs, represents the solution only by means of geometric primitives. Norato [Norato, 2018] provides an extensive description of works dealing with these approaches. The precursor of these methods is the bubble method (Eschenauer et al. [Eschenauer et al., 1994]), one of the first form of topology optimization. Still this approach requires re-meshing during the optimization process which affects its computational burden. Another work that can be consider as belonging to this category is the Feature-based topology optimization [Cheng et al., 2006], where the structure is obtained as the results of Boolean operations over simple geometric feature structures, in this case holes. The same author proposes in [Chen et al., 2008] an implicit control of the solution quality using a quadratic term of the energy in the formulation of the optimization problem. In the work of Seo et al. [Seo et al., 2010], the solution is described using trimmed spline surfaces and the isogeometric analysis. In the Adaptive Mask Overlay Topology Synthesis Method, Saxena et al. [Saxena, 2011] circular, rectangular or elliptical holes are considered in the solution and the structural analysis is performed on the remaining solid structure using hexagonal elements. A similar approach is applied by Wang et al. [Wang et al., 2012]: a fixed mesh is employed this time and a regularized Heaviside function is used to evaluate the mechanical properties on a fixed finite element mesh. Liu et al. [Liu et al., 2014] proposed to use Compactly supported Radial basis functions to interpolate simple geometric features in a level set topology optimization framework. The performance of the resulting structure was then computed thanks to the extended finite element method (XFEM). In the ISOCOMP approach of Lin et al. [Lin et al., 2015], the simultaneous optimization of both the location of holes and the layout of material is studied with the use of both isogeometric analysis and the Hierarchical Partition of Unity Field Compositions (HPFC) theory, which is employed for both geometry and solution field approximations.

In this second family of methods, four approaches represent a solution as the union of geometric entities that can move, stretch and analytically modify their shapes: the moving morphable components (MMC) method (Guo et al. [Guo et al., 2014, Guo et al., 2016], Zhang

et al. [Zhang et al., 2016c, Zhang et al., 2017d]) ; the geometry projection (GP) method (Bell et al. [Bell et al., 2012], Norato et al. [Norato, 2015], Zhang et al. [Zhang et al., 2016a]); the Method of Moving Morphable Bars (MMB) Hoang et al. [Hoang and Jang, 2017] and the Moving Node Approach (MNA), introduced in the Master thesis of Overvelde [Overvelde, 2012]. These approaches are also known as explicit topology optimization approaches, highlighting the fact that the material description is explicitly described, instead of implicitly as it is the case for direct density based approaches, where the layout is describe by a mapping of presence and absence of material. In [Guo et al., 2014], an explicit level set function is used to determine the geometry of moving morphable components with uniform thickness. The XFEM approach was employed as an alternative for the displacement computation [Kreissl and Maute, 2012, Van Miegroet and Duysinx, 2007, Wei et al., 2010]. The same framework is also presented in Zhang et al. [Zhang et al., 2016d] where MMC is identified as a way of extending techniques commonly employed in shape optimization, with the freedom inherited from topology optimization. Moving components with variable thickness are used in [Zhang et al., 2016c], and the ersatz material model is employed to enhance computational efficiency. In the work of Zhang et al. [Zhang et al., 2016b] length scale control is achieved by directly controlling the component's minimal thickness. In Deng et al. [Deng and Chen, 2016] the MMC approach is successfully implemented to solve various types of problems, like the design of compliant mechanisms and of low-frequency resonating micro devices. In the work of Zhang et al. [Zhang et al., 2017d], the MMC framework is extended to 3D structures. The complexity of the solution is controlled in [Zhang et al., 2017f] by controlling the maximum number of components in the final solutions.

The Moving Morphable Voids approach [Zhang et al., 2017c] makes use of B-splined shaped holes, explicitly piloted in the topology optimization. The proposed framework, not only reduces the optimization burden due to variables reduction, but also impacts the cost of the evaluation of the displacement field, eliminating the element completely contained in the void region from the analysis. The same framework is applied to tackle stress based topology optimization in [Zhang et al., 2018b]. In Zhang et al. [Zhang et al., 2017e] the control of the solution is achieved by varying explicitly the boundaries of Components or voids using B-spline curves. In Takalloozadeh et al. [Takalloozadeh and Yoon, 2017] the topological derivative is implemented in the MMC approach. In the work of [Guo et al., 2017] the ability of MMC to determined self-supported structures is studied and in Liu et al. [Liu et al., 2017] the MMCs/MMVs framework is employed to determine graded lattice structures achievable with additive layer manufacturing technologies. In Hou et al. [Hou et al., 2017] the MMC framework is proposed based on Isogeometric Analysis (IGA) instead of finite element analysis (FEA). In Zhang et al. [Zhang et al., 2018d], the MMC approach is employed to design multi-material structures and in Zhang et al. [Zhang et al., 2018c] it is employed to find the best layout of stiffening ribs including buckling constraints. Geometric non-linearities are considered in Zhu et al. [Zhu et al., 2018] for MMC and in Xue et al. [Xue et al., 2019] for MMV. In Lei et al. [Lei et al., 2019] machine learning techniques like: support vector regression (SVR) [Smola and Schölkopf, 2004] and the K-nearest-neighbors (KNN) [Altman, 1992], were adopted in order to speed up the resolution of the optimization problem, under MMC framework. In Liu et al. [Liu et al., 2018a] an efficient strategy is proposed to decouple the finite element mesh discretization from the discretization employed to assemble the stiffness

matrix on the basis of the geometric configuration. In Sun et al. [Sun et al., 2018] the topology optimization of a 3D multi-body systems considering large deformations and large overall motion is achieved using MMC.

With the Method of Moving Morphable Bars (MMB) Hoang et al. [Hoang and Jang, 2017] introduce round ended bars in the topology optimization framework that can overlap and change both shape and position. As main features of this work one can identify the control of both minimal and maximal thickness of the components, the use of sigmoid Heaviside function to project the components on the fixed finite element mesh, the use of SIMP material model to penalize intermediate density and the Boolean union of components realized making the product of Heaviside functions. MMB was also studied in Wang et al. [Wang et al., 2018] for the layout of planar multi-component systems.

The Geometry Projection (GP) approach was initially introduced by Norato et al. [Norato et al., 2004], for the shape optimization of holes over a fixed finite element mesh. This basic idea was developed further by Bell et al. [Bell et al., 2012] to make the topology optimization of structures composed by rectangular components in 2D, by Norato et al. [Norato et al., 2015] using 2D round ended bar components that can both overlap and merge. The same author proposed to use this approach to find the best distribution of short fiber reinforcement in [Norato, 2015]. In Zhang et al. [Zhang et al., 2016a] Geometry projection approach was implemented for 3D solid structures composed of rectangular plates. In Zhang et al. [Zhang et al., 2017a] stress based topology optimization is conducted on 2D topology optimization problems, using GP. The design of unit cells for lattice materials in 3D is considered in Watts et al. [Watts and Tortorelli, 2017]. In Zhang et al. [Zhang et al., 2018a] curved plates are considered as building blocks of 3D topology optimization problems using GP. In Zhang et al. [Zhang and Norato, 2017] Geometry Projection was employed to design the rib reinforcement of plates in 3D. In the work of Norato [Norato, 2018] supershapes geometric features are employed as building blocks in a 2D topology optimization framework based on geometry projection. In Kazemi et al. [Kazemi et al., 2018], the Geometry Projection was applied to multi-material design of 2D and 3D structures. In Zhang et al. [Zhang and Norato, 2018] a tunneling strategy is proposed to alleviate the initial point dependency in Geometry projection.

This chapter focused on the following three frameworks, which follow different approaches for achieving explicit topology optimization of 2D and 3D structures: the Moving Morphable Components (MMC) with ersatz material approach, the Geometry Projection (GP) approach and Moving Node Approach (MNA). In this chapter it is also proposed a new general framework that was named Generalized Geometry Projection (GGP), of which all three approaches can be seen as particular cases. A further contribution resides in the proposal of a specific saturation strategy that can be adopted to overcome the difficulties that can be encountered in the assembly of geometric features, required in all considered explicit topology optimization approaches. Finally, a root cause of optimization convergence difficulties is extensively analyzed and possible solutions to circumvent this issue are discussed.

The remainder of this chapter is organized as follows: Section 3.1 introduces a 2D demonstrator framework for all reviewed Lagrangian techniques. Sections 3.2, 3.3 and 3.4 review respectively Moving Morphable Components with ersatz material model, Geometry Projection

and Moving Node Approach. In section 3.5 a proposed strategy called Generalized Geometry Projection is introduced and reviewed strategies are shown to be particular subcases of this strategy. In section 3.6 the most common geometric assembly strategies employed in this kind of framework are reviewed. In section 3.7 the analytic evaluation of sensitivities is provided. A 2D investigation of the aforementioned strategies is proposed in section 3.8. Then an application to stress based topology optimization is conducted in section 3.9. Finally MNA is applied to the 3D topology optimization of engine pylon and mounts in section 3.10.

### 3.1 2D demonstration framework

This section first describes three of the existing 2D explicit topology optimization approaches: Moving Morphable Components (MMC) with ersatz material model; Geometry Projection (GP); Moving Node Approach (MNA). Then the proposed Generalized Geometry Projection (GGP) is described. This is aimed at unifying into a single formulation the other reviewed existing approaches. To simplify the equations the same geometric components proposed in Norato et al. [Norato, 2015] will be considered for all reviewed approaches c.f. figure 3.1a. This component is defined by five geometric parameters defining the position of the center of the component  $\{X, Y\}$ , the components' dimensions  $\{L, h\}$  and the component's orientation  $\{\theta\}$ . The final topology will then be described as a superposition of such basic components. In this section  $\omega$  indicates the area occupied by a geometric component and  $\partial\omega$  denotes its boundary. If more than one component is used to describe the area occupied by material, then  $\omega_i$  and to  $\partial\omega_i$  refer to the  $i^{th}$  component's area and boundary respectively. In such case  $\omega$  and to  $\partial\omega$  refer to area and boundary of the union of all components  $i$  ( $\omega = \cup_{i=1}^n \omega_i$ ). In order to

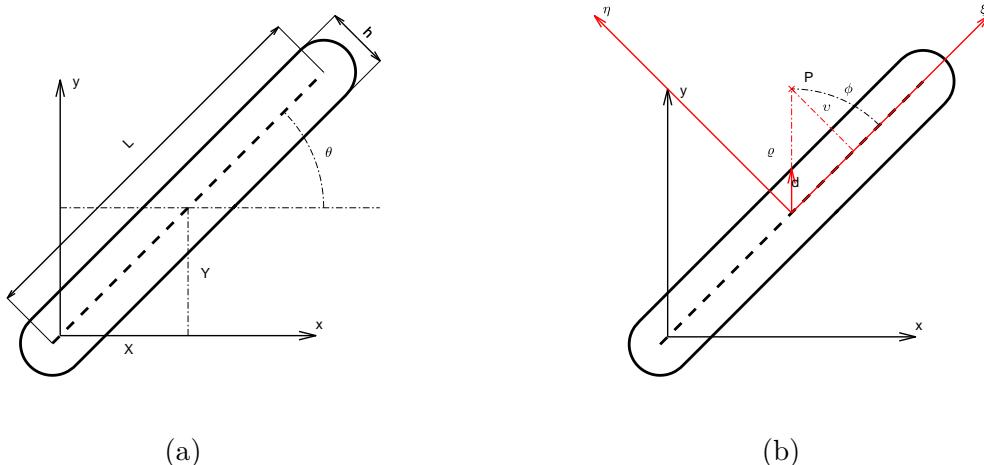


Figure 3.1: Geometric primitive (i.e. basic geometric component) used for all the reviewed methods. (a) The explicit geometric parameters associated to the component are  $\{X, Y, L, h, \theta\}$ . (b) Plot defining the local polar coordinates  $\{\rho, \phi\}^T$ , the distance from the component boundary  $d$ , and the distance from the component middle axis  $v$

define explicitly the contour of this component, polar coordinates  $\{\varrho, \phi\}^T$  are introduced as defined in figure 3.1b. Here  $d$  denotes the radial distance of the component's center  $\{X, Y\}^T$  from the component boundary  $\partial\omega$  that can be computed as a function of the angle  $\phi$  and of the component sizes  $L$  and  $h$  as:

$$d(\phi, L, h) = \begin{cases} \sqrt{\frac{h^2}{4} - \frac{L^2 \sin \phi^2}{4}} + \frac{L}{2} |\cos \phi| & \text{if } \cos \phi^2 \geq \frac{L^2}{L^2+h^2}, \\ \frac{h}{2|\sin \phi|} & \text{otherwise.} \end{cases} \quad (3.1)$$

This piece-wise definition is at least of class  $\mathbf{C}^1(\mathbb{R})$ . Given a point  $\{X_g\} \equiv \{x, y\}^T \in \mathbb{R}^2$ , its polar coordinates can be defined as:

$$\varrho(x, y, X, Y) = \sqrt{(x - X)^2 + (y - Y)^2} \quad (3.2)$$

$$\phi(x, y, X, Y, \theta) = \begin{cases} \arctan\left(\frac{y-Y}{x-X}\right) - \theta & \text{if } x \neq X, \\ \frac{\pi}{2}\text{sign}(y - Y) - \theta & \text{if } x = X. \end{cases} \quad (3.3)$$

A consequence of the above definitions:

$$\{X_g\} \in \omega \cup \partial\omega \Leftrightarrow d(\phi, L, h) \geq \varrho(x, y, X, Y) \quad (3.4)$$

Another way of describing the same component is by the use of the signed distance as described in Norato et al. [Norato, 2015]:

$$\varsigma(v, h) := v - \frac{h}{2} \quad (3.5)$$

where  $v$  is the distance from a point  $\{X_g\} \in \mathbb{R}^2$  to the component medial axis, which can be further expressed as a function of  $\{X_g\}$ 's polar coordinates:

$$v(\varrho, \phi, L, h) = \begin{cases} \sqrt{\varrho^2 + \frac{L^2}{4} - \varrho L |\cos \phi|} & \text{if } \varrho^2 \cos \phi^2 \geq \frac{L^2}{4}, \\ \varrho |\sin \phi| & \text{otherwise} \end{cases} \quad (3.6)$$

Accordingly,

$$\{X_g\} \in \omega \cup \partial\omega \Leftrightarrow \varsigma(v, h) \leq 0 \quad (3.7)$$

The structural model used during the topology optimization will then be described by  $n$  components, each involving following six<sup>1</sup> design variables

$\{x_i\} = \{X_i, Y_i, L_i, h_i, \theta_i, m_i\}^T$ . For the whole structure involving  $n$  components the structural performance will thus depend on  $6n$  design variables denoted as vector  $\{x\}$ :

$$\{x\} = \left\{ \begin{array}{c} \{x_1\} \\ \{x_2\} \\ \vdots \\ \{x_n\} \end{array} \right\} \quad (3.8)$$

---

<sup>1</sup>additional variable  $m_i$  is introduced in the geometry projection approach to make a component vanish in the same way as it is done in density based approaches.

Each of the reviewed methods has its own way of updating the structural model that is used to evaluate the performance of a given design. A common feature is the fact that a linear elastic-static finite element model is employed. Here a structured uniform mesh based on  $dx \times dx$  solid elements in plane stress is adopted, where  $dx$  is the element  $x$  and  $y$  size. The Poisson's ratio  $\nu$  and the thickness  $t$  in the out of plane direction is considered to be a constant over all the design domain  $D$ . The Young's modulus  $E$  depends on the fact that the considered point belongs or not to  $\omega$ . A common assumption is that the Young's modulus is piece-wise uniform over each element. For this reason, the elementary stiffness matrices  $[K^{el}]$  are all of the same form i.e:

$$[K^{el}] = E^{el} [K_0] \quad el = 1, 2, \dots, N \quad (3.9)$$

where  $E^{el}$  is the Young's modulus of the  $el^{th}$  element,  $[K_0]$  is the  $8 \times 8$  stiffness matrix of a  $dx \times dx$  2D solid element in plane stress condition with thickness  $t = 1$  and  $N$  is the total number of elements in the structural model. Using classic finite element theory, one can assemble the global stiffness matrix  $[K]$ :

$$[K] = \bigoplus_{el=1}^N [K^{el}] \quad (3.10)$$

This is used to write the static balance equation in terms of the free degrees of freedoms  $\{U\}$  as:

$$[K]\{U\} = \{F\} \quad (3.11)$$

where  $\{F\}$  is the external loads vector. If the boundary conditions are at least isostatic and  $E^{el} \geq E_{min} > 0$ , then the system of equation  $[K]$  is non-singular and the system of equation (3.11) can be solved to find the displacement vector  $\{U\}$ .

$$\{U\} = [K]^{-1}\{F\} \quad (3.12)$$

In many topology optimization formulations the mass of the structure is also of interest, which is usually considered in terms of the volume fraction  $V$  associated to a given structural design layout (since  $V$  is proportional to the mass of a solution for a homogeneous material). To compute this volume fraction  $V$  the local density  $\rho^{el}$  are determined based on the design vector  $\{x\}$  as will be presented in details in the next subsections for each approach.

An example of a topology optimization formulation within this framework is provided in eq. (3.13). The classical formulation consists in minimizing the compliance of the structure (i.e. maximizing its stiffness) subjected to a volume fraction constraint and design space constraints:

$$\begin{cases} \min_{\{x\}} C = \{U\}^T \{F\} \\ s.t. \\ V = \frac{\sum_{el=1}^N \rho^{el}}{N} \leq V_0 \\ \{l_b\} \leq \{x\} \leq \{u_b\} \end{cases} \quad (3.13)$$

Where  $\{l_b\}$  and  $\{u_b\}$  are respectively the vectors of lower bound and upper bound for the

design variables and  $V_0$  is the maximum allowed volume fraction in the final solution.

### 3.2 Moving Morphable Components (MMC) with ersatz material model

This section reviews the moving morphable components (MMC) method (Guo et al. [Guo et al., 2014, Guo et al., 2016]). MMC inherits from the level set method [Allaire et al., 2004] a Topology Description Function (TDF), that is positive inside the area occupied by the union of all components  $\omega$ , equal to zero on the boundary  $\partial\omega$  of the component and negative outside the component. The TDF values are then used either to apply the XFEM [Wei et al., 2010] on the boundaries of the components either to compute the element stiffness matrix using ersatz material model [Zhang et al., 2016c]. In this paper will focus on the application of the latter. The reader must note that some minor modifications to the original formulation of [Zhang et al., 2016c] were made in order to include the round ended bar components (cf. Fig. 3.1a) with uniform thickness. The structural topology description is obtained using a topology description function (TDF), denoted  $\chi$ , that satisfies following relations:

$$\begin{cases} \chi > 0 & \text{if } \{X_g\} \in \omega, \\ \chi = 0 & \text{if } \{X_g\} \in \partial\omega, \\ \chi < 0 & \text{if } \{X_g\} \in D \setminus \omega. \end{cases} \quad (3.14)$$

where  $D$  represents the total area of the design domain (full and voids). When more than one component is considered, the TDF of each component, denoted as  $\chi_i = \chi_i(\{X_g\})$ , is defined  $\forall i = 1, \dots, n$  such that:

$$\begin{cases} \chi_i > 0 & \text{if } \{X_g\} \in \omega_i, \\ \chi_i = 0 & \text{if } \{X_g\} \in \partial\omega_i, \\ \chi_i < 0 & \text{if } \{X_g\} \in D \setminus \omega_i. \end{cases} \quad (3.15)$$

Since  $\omega = \cup_{i=1}^n \omega_i$  one can verify that  $\chi = \max_i \chi_i$  satisfies the conditions (3.14). Given a component described by the geometric variables  $\{X_i, Y_i, L_i, h_i, \theta_i\}$ , several choices are available for defining the TDF  $\chi_i(\{X_g\})$ . Here it was considered the following relationship, which has the advantage of allowing simple derivations:

$$\chi_i = 1 - \left( \frac{4v_i^2}{h_i^2} \right)^\alpha \quad \text{with } \alpha \geq 1 \quad (3.16)$$

Figure 3.2 illustrates the contour plot of the TDF  $\chi_i$  contour of the generic component of figure 3.1a.

The presence or absence of material in the design under consideration can be obtained by the Heaviside function  $H(x)$ , applied to the topology description function of the union of all the components  $\chi$ . In order to have a regular behavior of the optimization problem responses,

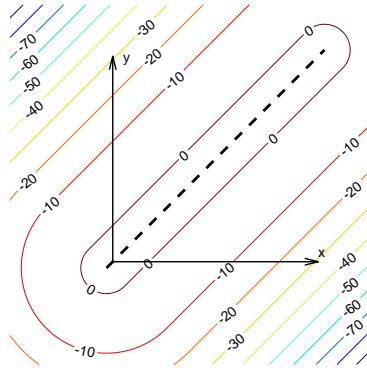


Figure 3.2:  $\chi(\{X_g\})$  contour plot of the generic component of figure 3.1a.  $\alpha = 1$ ,  $X = 1$ ,  $Y = 1$ ,  $L = 3$ ,  $h = 0.5$ ,  $\theta = \frac{\pi}{4}$  were considered. The domain  $D$  of the plot is  $[-1, 2.5] \times [-1, 2.5]$

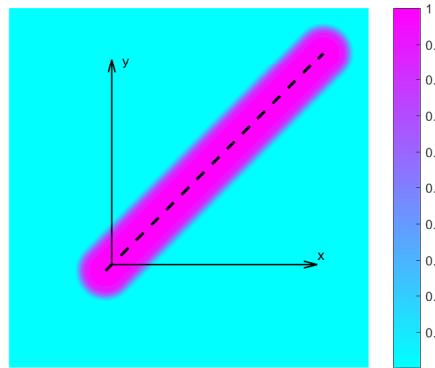


Figure 3.3:  $H_\epsilon(\chi(\{X_g\}))$  filled contour plot of the component in figure 3.1a.  $\alpha = 1$ ,  $X = 1$ ,  $Y = 1$ ,  $L = 3$ ,  $h = 0.5$ ,  $\theta = \frac{\pi}{4}$ ,  $\beta = 0.01$ ,  $\epsilon = 0.6$  were considered. The domain  $D$  of the plot is  $[-1, 2.5] \times [-1, 2.5]$

the Heaviside function  $H(x)$  is replaced by a regularized version  $H_\epsilon(x)$ :

$$H_\epsilon(x) = \begin{cases} 1, & \text{if } x > \epsilon, \\ \frac{3(1-\beta)}{4} \left( \frac{x}{\epsilon} - \frac{x^3}{3\epsilon^3} \right) + \frac{1+\beta}{2} & \text{if } -\epsilon \leq x \leq \epsilon, \\ \beta & \text{otherwise.} \end{cases} \quad (3.17)$$

Where  $0 < \epsilon < 1$  is a parameter that controls the amplitude of the transition between the minimum value of  $\beta$  ( $0 < \beta \ll 1$ ) and 1. The same generic component is used in figure 3.3 to plot  $H_\epsilon(\chi(\{X_g\}))$ . One can observe that the smooth variation, from a value of 1 inside the component to a value of  $\beta$  outside it, is localized in a small transition zone, denoted  $D_g$ ,

that can be determined using equations (3.16)-(3.17):

$$D_g^{MMC} = \left\{ \{X_g\} \mid \frac{h}{2}(1-\epsilon)^{\frac{1}{2\alpha}} \leq v \leq \frac{h}{2}(1+\epsilon)^{\frac{1}{2\alpha}} \right\} \quad (3.18)$$

The width of this transition zone is denoted by  $w_g$ :

$$w_g^{MMC} = \frac{h}{2} \left[ (1+\epsilon)^{\frac{1}{2\alpha}} - (1-\epsilon)^{\frac{1}{2\alpha}} \right] \quad (3.19)$$

A peculiarity of MMC is that the width of the transition zone is directly proportional to the component's thickness  $h$ . A direct consequence is that smaller components will have faster variation between full material and voids. The effect of this behavior on the ill conditioning of the optimization problem will be investigated on numerical examples in the implementation section.

According to [Guo et al., 2005], the value of the Young's modulus in the  $el^{th}$ -element is considered to be:

$$E^{el} = \frac{E \left( \sum_{j=1}^4 (H_\epsilon(\chi_j^{el}))^q \right)}{4} \quad (3.20)$$

Where  $\chi_i^{el}, i = 1, \dots, 4$  are the values of the TDF at the four nodes of the element  $el$  and  $q$  is a parameter that has the role of penalization, in order to render the variation of Young's modulus even faster at the boundary of the component. In figure 3.4a, the single component example of figure 3.1a has been used to plot the distribution of Young's modulus according to equation (3.20) over a  $50 \times 50$  finite element mesh. In order to obtain the local density  $\rho^{el}$ , which was not explicitly considered in [Zhang et al., 2016c], an equivalent expression that leads to the same value of volume fraction for the same configuration is proposed here:

$$\rho^{el} = \frac{\sum_{j=1}^4 (H_\epsilon(\chi_j^{el}))}{4} \quad (3.21)$$

In figure 3.4c the corresponding distribution of  $\rho^{el}$  is also represented. Figures 3.4c and 3.4a look very similar except for the transition on element boundary that is faster for the Young's modulus plot due to the effect of the penalization parameter  $q$ .

### 3.3 Geometry Projection (GP)

This section reviews the approach proposed by Norato et al. [Norato, 2015]. Geometry projection first computes the signed distance between each element central point and each component surface. The element local volume fraction is then computed by the mean of a spherical sampling window centered in the element centroid. Density and Young's modulus in the element are computed as function of the volume fraction of the sampling window that is occupied by material. The density coming from each component is unified using the maximum function or its smooth approximation [Kreisselmeier and Steinhauser, 1980]. The solution is described by the union of geometric primitives such as the one in figure 3.1a. To update

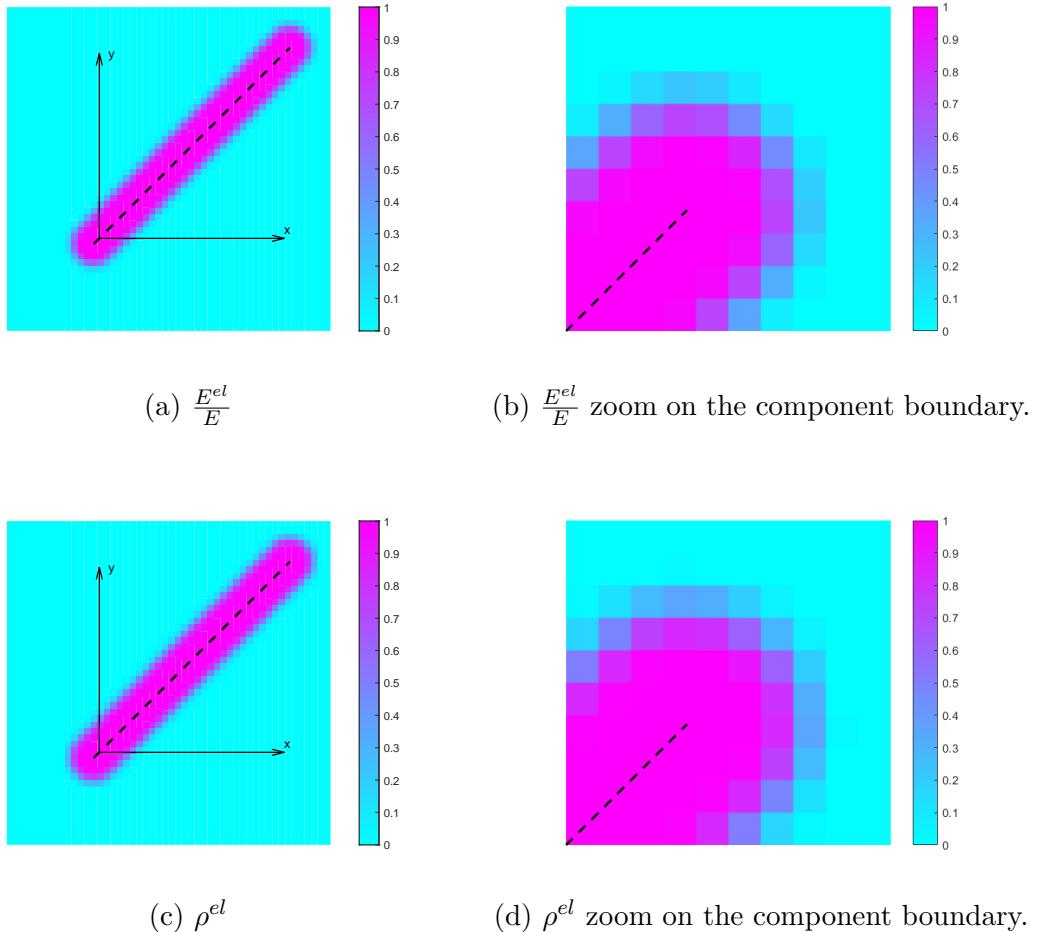


Figure 3.4: Distribution of  $\frac{E^{el}}{E}$  (a-b) and  $\rho^{el}$  (c-d) for the generic component of figure 3.1a and for a  $50 \times 50$  FE mesh over the domain of  $X_g$ . The same parameters of figure 3.3 are also considered here with  $q = 2$ .

model densities the geometry projection method is employed [Norato et al., 2004]. A circular sampling window  $\mathbf{B}_P^r$  of radius  $r$  is considered around the  $el^{th}$ -element center  $\{X_g^{el}\}$  c.f. figure 3.5. The local volume fraction  $\delta_i^{el}$  is simply given by the fraction of the window that is filled with material:

$$\delta_i^{el} = \frac{|\mathbf{B}_P^r \cap \omega_i|}{|\mathbf{B}_P^r|} \quad (3.22)$$

where  $|\cdot|$  denotes the measure of the area. The denominator can be computed analytically as the area of the circle  $|\mathbf{B}_P^r| = \pi r^2$ . The numerator of equation (3.22) on the other hand is more complex but can be approximately computed with the assumption that  $r$  is small enough. In this case the restriction of  $\partial\omega$  to the circle  $\mathbf{B}_P^r$  can be considered as a straight

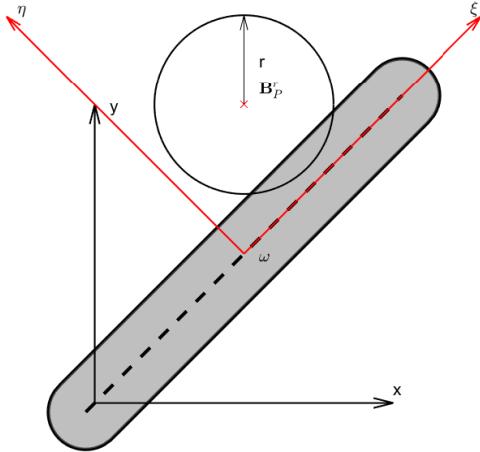


Figure 3.5: Basic component and notations associated to the Geometry Projection method [Norato, 2015]

line. As a consequence one can compute:

$$\delta_i^{el} \approx \begin{cases} 0 & \text{if } \varsigma > r, \\ \frac{1}{\pi r^2} \left[ r^2 \arccos \left( \frac{\varsigma}{r} \right) - \varsigma \sqrt{r^2 - \varsigma^2} \right] & \text{if } -r \leq \varsigma \leq r, \\ 1 & \text{otherwise.} \end{cases} \quad (3.23)$$

Where the signed distance  $\varsigma$  (cf. eq. (3.5)) is computed in the  $el^{th}$  element centroid  $\{X_g^{el}\}$  and with respect to the  $i^{th}$  component. In order to avoid stiffness matrix singularities, the local densities are modified as follows.

$$\tilde{\delta}_i^{el} = \delta_{min} + (1 - \delta_{min})\delta_i^{el} \quad (3.24)$$

where  $\delta_{min}$  is the minimum of local volume fraction to be considered in the analysis. Moreover:

$$\hat{\delta}_i^{el}(m_i, \gamma) = \tilde{\delta}_i^{el} m_i^\gamma \quad (3.25)$$

Where  $m_i$  is the  $i^{th}$  component mass or out of plane thickness [Norato, 2015] and  $\gamma \geq 1$  penalizes the intermediate value of the component's mass. The local densities are finally computed by taking the union of all the components using a smooth approximation of the maximum function:

$$\rho^{el}(\gamma_v, \kappa) = \Pi(\{\hat{\delta}^{el}(\{m\}, \gamma_v)\}, \kappa) \quad (3.26)$$

where  $\kappa$  is an aggregation constant and  $\{\hat{\delta}^{el}(\{m\}, \gamma_v)\}$  is the vector of local density stemming from each component. Here the form of the smooth approximation of the maximum function  $\Pi$  is not specified, this will be investigated in details in the section 3.6. In order to determine the value of the Young's modulus of an element, the following equation is used which involves a second penalty parameter  $\gamma_c > \gamma_v$

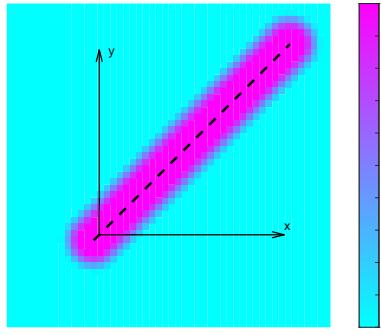


Figure 3.6: Distribution of  $\frac{E^{el}}{E}$  for the generic component of figure 3.1a and for a  $50 \times 50$  mesh over the domain of  $X_g$ .  $X = 1, Y = 1, L = 3, h = 0.5, \theta = \frac{\pi}{4}, \gamma_c = 3, r = 0.105, \delta_{min} = 10^{-6}$  were considered. Due to the choice of  $m = 1$  the same distribution is obtained for  $\rho^{el}$ .

$$E^{el} = \rho^{el}(\gamma_c, \kappa)E \quad (3.27)$$

This penalization is very similar to the one adopted by the SIMP approach and is effective in order to progressively eliminate a component with an intermediate value of  $m_i$  throughout the optimization iterations [Norato, 2015]. In figure 3.6, the generic component of figure 3.1a is considered to show the distribution of both Young's modulus and element densities over a  $50 \times 50$  mesh. Note again that the transition zone from full material to void is concentrated on the boundary of the component:

$$D_g^{GP} = \left\{ \{X_g\} \mid \frac{h}{2} - r \leq v \leq \frac{h}{2} + r \right\} \quad (3.28)$$

As a consequence this time the thickness of the transition zone is:

$$w_g^{GP} = 2r \quad (3.29)$$

This thickness does not depend on the component size  $h$ . In order to achieve differentiable density distributions,  $h$  will need to be greater or equal than  $2r$ . To delete inactive components, variable  $m$  can still be used within the optimization algorithm.

### 3.4 Moving Node Approach (MNA)

Overvelde [Overvelde, 2012], proposed in his master's thesis an alternative flow-inspired topology optimization approach, the Moving Node Approach (MNA). For this approach, the building blocks of a solution are defined as mass nodes. Each element's center position is recomputed with respect to a local coordinate system in each component center. Then weighting functions are directly applied to the local variable to compute the component local density contribution. In order to compute the union between the components, this time, the densities are summed. Since the sum can be greater than one, to keep the resulting density bounded

from 1 an adapted procedure was proposed in [Overvelde, 2012] called asymptotic density. Another peculiarity of Overvelde's work was the fact of using either Finite Element Analysis or meshless method (cf. [Nguyen et al., 2008]) for the displacement evaluation. The idea was to reduce both design variable and degrees of freedom number, using the mass nodes for both the geometric description and the solution displacement. Unfortunately in [Overvelde, 2012] the gain in DOFs number was compensated by the stiffness matrix cost (both in memory and elapsed time) and by its ill conditioning where the mass nodes reached each other. The update of the finite element model is done by operating through weighting functions  $w$  that are driven by the geometry of each component. This work considered a modified weighting function with respect to [Overvelde, 2012] in order to consider round ended bar components. For the mass node of figure 3.1a one can write:

$$w(v, h, \varepsilon) = \begin{cases} 1 & \text{if } v \leq l, \\ a_3v^3 + a_2v^2 + a_1v + a_0 & \text{if } l < v < u, \\ 0 & \text{otherwise.} \end{cases} \quad (3.30)$$

Where

$$l = \frac{h}{2} - \frac{\varepsilon}{2} \quad (3.31)$$

$$u = \frac{h}{2} + \frac{\varepsilon}{2} \quad (3.32)$$

$$a_3 = \frac{2}{\varepsilon^3} \quad (3.33)$$

$$a_2 = -\frac{3h}{\varepsilon^3} \quad (3.34)$$

$$a_1 = 3 \frac{(h^2 - \varepsilon^2)}{\varepsilon^3} \quad (3.35)$$

$$a_0 = -\frac{(h + \varepsilon)^2(h - 2\varepsilon)}{4\varepsilon^3} \quad (3.36)$$

The local density can then be computed as:

$$\delta_i^{el} = m_i^\gamma w(v_i^{el}, h_i, \varepsilon_i) = m_i^\gamma w_i^{el} \quad (3.37)$$

Where  $v_i^{el}$  refers to the distance from the  $el^{th}$  element centroid  $\{X_g^{el}\}$  to the  $i^{th}$  component middle axis computed using equation (3.6). To make the union of all mass nodes an asymptotic density was employed. Here this approach was generalized using a general notation for the union:

$$\rho^{el} = \Pi(\{\delta\}_v^{el}, \kappa) \quad (3.38)$$

Finally the Young's modulus is updated using a power law:

$$E^{el} = E_{min} + (E - E_{min})(\Pi(\{\delta\}_c^{el}, \kappa))^{p_b} \quad (3.39)$$

Where  $p_b \geq 1$  is used to penalize intermediate densities. In figures 3.7a,3.7c both Young's modulus distribution and densities are considered for the same generic component and same

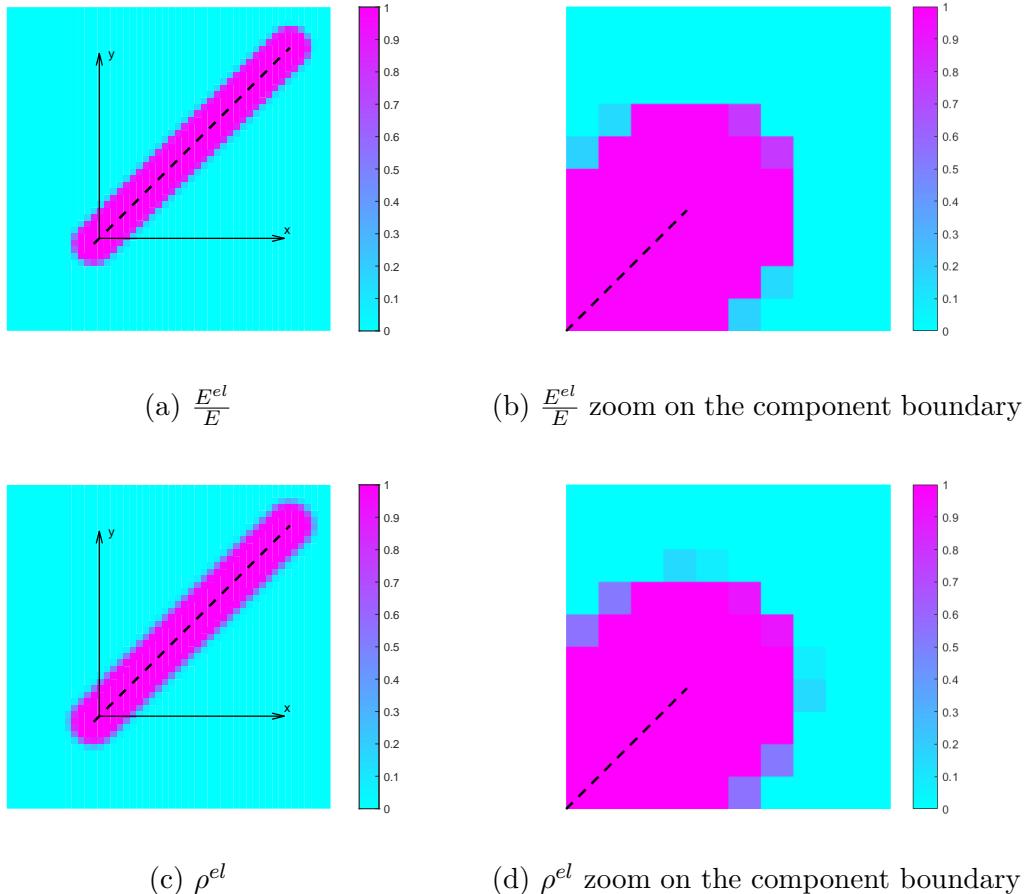


Figure 3.7: Distribution of  $\frac{E^{el}}{E}$  (a-b) and  $\rho^{el}$  (c-d) for the generic component of figure 3.1a and for a  $50 \times 50$  FE mesh over the domain of  $X_g$ .  $X = 1, Y = 1, L = 3, h = 0.5, \theta = \frac{\pi}{4}, \gamma = 3, \varepsilon = 0.14, p_b = 3, E_{min} = 10^{-6}$  were considered.

mesh as for figures 3.4a, 3.4c and 3.6. Note that in this case as well the gray region is localized in the transition zone defined this time by:

$$D_g^{MNA} = \left\{ \{X_g\} \mid \frac{h - \varepsilon}{2} \leq v \leq \frac{h + \varepsilon}{2} \right\} \quad (3.40)$$

The thickness of this transition zone is then defined as:

$$w_g^{MNA} = \varepsilon \quad (3.41)$$

This thickness is, as for Geometry Projection, independent from the component thickness  $h$ . On the other hand one can observe in figure 3.7a, compared to Geometry Projection, the effect of the penalty  $p_b > 1$  that reduces the value of the Young Modulus in the transition zone.

## 3.5 Generalized Geometry Projection

This subsection introduces the proposed Generalized Geometry Projection method as a generalization of the Geometry Projection (Bell et al. [Bell et al., 2012]; Norato et al. [Norato, 2015]; Zhang et al. [Zhang et al., 2016a]). Moreover it will be shown that the proposed approach can recover all the reviewed approaches in terms of relationships between the geometric configuration and finite element model update. Essentially, all reviewed approaches can be seen as a particular case of the proposed Generalized Geometry Projection method. Let us first formalize the general procedure that is common to all existing explicit approaches c.f. fig3.8. The first step consists in choosing the geometric primitives, i.e. the building

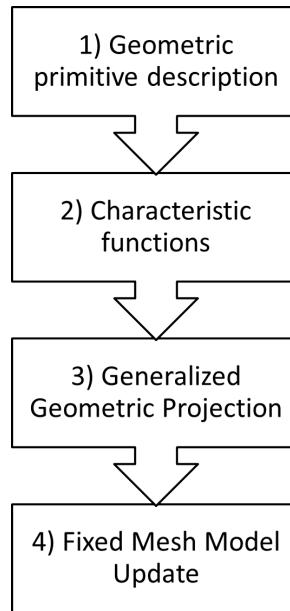


Figure 3.8: General procedure employed by explicit approaches. In step 1) Geometric primitives are chosen, so that their layout, shape and sizes can be explicitly driven by the optimization procedure. In step 2) According to the geometry description characteristic functions has to be defined for each feature and for all material properties. These will be computed in each sampling window Gauss points. In step 3) Generalized Geometry Projection is used to compute the value of local volume fraction for all material properties. Finally in step 4) the Finite element model is updated according to the value of each local volume fraction in each element centroid.

blocks that are going to be used to build the solution through Boolean operations. As for all reviewed approaches, round ended bar components (cf. fig. 3.1a) in a 2D design space will be considered here as geometric primitive. Then, characteristic functions  $\Upsilon$  have to be defined for each geometric primitive  $i$ . A characteristic function can be defined for the set of points inside the geometric primitive  $\omega_i$  as:

$$\Upsilon(\{X_g\}, \omega_i) = \begin{cases} 1 & \text{if } \{X_g\} \in \omega_i \\ 0 & \text{otherwise.} \end{cases} \quad (3.42)$$

For implementation purpose (in order to improve the regularity of the functions in the optimization problem),  $W_i(\{X_g\}, \{X_i\}, \{r\})$  should be chosen to be a regular approximation of  $\Upsilon(\{X_g\}, \omega_i)$ . Accordingly, for the choice of a characteristic function it is required here :

$$\begin{cases} 0 \leq W_i(\{X_g\}, \{X_i\}, \{r\}) \leq 1 \\ \lim_{r \rightarrow \lambda} (W_i(\{X_g\}, \{X_i\}, \{r\})) = \Upsilon(\{X_g\}, \omega_i) \\ W_i(\{X_g\}, \{X_i\}, \{r\}) \in \mathbf{C}^1(\mathbb{R}^{d_g}) \end{cases} \quad (3.43)$$

where  $d_g$  is the dimension of the  $\{X_g\}$  space (in the present case  $d_g = 2$ , since only 2D problems are considered) and  $\lambda \in \{0, +\infty\}$ . The vector of hyper-parameters  $\{r\}$  will control the length scale of the transition of  $W_i$  between 0 and 1. The functions  $W_i$  will need to be non-increasing with respect to any direction that points outward of the component. For a more formal definition, the following procedure is introduced: given a point  $\{X_g\} \in \mathbb{R}^{d_g}$ , one can find its projection on the geometric feature boundary as:  $\{X_g^*\} = \arg \min_{\{x\} \in \partial \omega_i} \|\{x\} - \{X_g\}\|$ . One can then find the outward direction in  $\{X_g^*\}$  defined as  $\{n^*\}$ . Finally for any  $\{X_g\}$  it is required that:

$$\left\{ \frac{\partial W_i(\{X_g\}, \{X_i\}, \{r\})}{\partial X_g} \right\}^T \{n^*\} \leq 0 \quad (3.44)$$

This condition avoids useless difficulties in optimization due to oscillations in the Heaviside function of each component.

As aforementioned, a great virtue of all reviewed approaches consists in using a unique finite element model to simulate each configuration, thus avoiding re-meshing. The proposed approach also proposes a procedure to update a given finite element model based on the configuration of the various components. In order to do so, the third step of Fig. 3.8 consists in using a procedure that transforms the continuous distribution of material represented by  $W_i(\{X_g\}, \{X_i\}, \{r\})$  into a piece-wise uniform distribution of Young's modulus and density inside each element of the FE mesh. The geometry projection proposed by Norato et al. [Norato et al., 2004] is here generalized to consider several sampling window shapes.<sup>2</sup>

For this purpose let's consider the following definition of the sampling window:

$$\mathbf{D}(\{X_g\}, p, R) = \{\{X\} \in \mathbb{R}^{d_g} \mid \|\{X\} - \{X_g\}\|_p \leq R\} \quad (3.45)$$

Next, the local density is computed in each element of the FE mesh using the formulation proposed in the generalized geometry projection method:

$$\delta_i^{el}(W_i, p, R) = \frac{\int_{\mathbf{D}(\{X_g^{el}\}, p, R)} W_i(\{X\}, \{X_i\}, \{r\}) d\Omega}{\int_{\mathbf{D}(\{X_g^{el}\}, p, R)} d\Omega} \quad (3.46)$$

This formulation can be seen as a weighted volume fraction estimation over the sampling

<sup>2</sup>Here only  $dx \times dx$  uniform meshes are studied, but the presented framework is also valid for non-uniform and irregular meshes. Moreover, note that the sampling window shape can eventually be shaped as the finite element mesh considering a slightly different formula in the sampling window definition that is not detailed here for conciseness.

window.

The evaluation of this expression can be done for example by Gauss quadrature:

$$\delta_i^{el} \approx \frac{\sum_{k=1}^{N_{gp}} \psi_k W_{ik}}{\sum_{k=1}^{N_{gp}} \psi_k} \quad (3.47)$$

where  $W_{ik}$  are the values of characteristic functions in Gauss point locations and  $\psi_k$  are the integration weights.<sup>3</sup> One must note that the characteristic function has not to be the same for Young's Modulus and density models.  $\{\delta^{el}\}_v$  refers to the vector of local volume fractions computed in the  $el^{th}$  element centroids for each component in the optimization using a density characteristic function denoted as  $W^v$ . In the same way  $\{\delta^{el}\}_c$  refers to the vector of local volume fraction computed in the  $el^{th}$  element centroid for each component in the optimization using density characteristic function denoted as  $W^c$ . Finally the last step in the general procedure of figure 3.8 consists in updating the fixed mesh of the FE model using:

$$E^{el} = \mathbb{M}(\{\delta^{el}\}_c, E, E_{min}, \kappa) \quad (3.48)$$

$$\rho^{el} = \mathbb{V}(\{\delta^{el}\}_v, \kappa) \quad (3.49)$$

Where  $\mathbb{M}$  and  $\mathbb{V}$  are regular functions that link respectively the Young's modulus and local densities in each finite element to the local volume fraction values  $\delta^{el}$  stemming from each geometric primitive.<sup>4</sup> It will now be shown that all three reviewed approaches can be recovered as a particular case by the Generalized Geometry Projection approach.

Let's consider the example of MMC with ersatz material model<sup>5</sup>. Let's consider  $W^v = H(\chi), W^c = (H(\chi))^q$ ,  $p \rightarrow \infty$  and  $R = \frac{\sqrt{3}}{2}dx$ . Let's consider Gauss-Legendre numerical integration in the sampling window, that for  $p \rightarrow \infty$ , is  $D \equiv [-R, R] \times [-R, R]$ ,  $2 \times 2$  Gauss points for the numerical evaluation of the integral of equation (3.46). For the function providing the element's Young's modulus it will be considered:

$$\mathbb{M}(\delta^{el}, E, E_{min}) = \lim_{p \rightarrow \infty} \delta^{el}(W^c, p, R) E_0 \quad (3.50)$$

Given these assumptions it is possible to obtain the volume over a sampling window as:

$$\lim_{p \rightarrow \infty} \int_{\mathbf{D}(\{X_g\}, p, \frac{\sqrt{3}}{2}dx)} d\Omega = \sum_{j=1}^4 \left( \frac{\sqrt{3}}{2}dx \right)^2 = 3dx^2 \quad (3.51)$$

---

<sup>3</sup>In [Wein and Stingl, 2018] a numerical integration of a smooth Heaviside function over the finite element was proposed. This work considering general shaped sampling window and also general radius can also be considered as a natural extension of his work.

<sup>4</sup>As a special case one could assemble geometric primitives before computing the local volume fractions. In this case the vectors of local volume fraction reduces to scalars computed that are unchanged by geometric assembly.

<sup>5</sup>This demonstration only applies to the case of  $dx \times dx$  uniform meshes. The same demonstration can be easily extended to  $dx \times dy$  uniform meshes simply changing sampling window definition. For the general situation of non-uniform irregular meshes, to recover the MMC formulation one should define local sampling window shapes and a more elastic numerical integration scheme based on triangulation.

Furthermore, given the choice of 4 Gauss points, the integration involved in the calculation of the local volume fraction of eq. (3.46) becomes:

$$\lim_{p \rightarrow \infty} \delta^{el}(W^c, p, R) \approx \frac{\frac{3}{4} dx^2 \sum_{j=1}^4 W^c(\mathbf{x}_j)}{3dx^2} = \frac{\sum_{j=1}^4 (H(\chi(\mathbf{x}_j)))^q}{4} \quad (3.52)$$

Accordingly the element's density can be expressed as:

$$\rho^{el} = \mathbb{V}(\delta^{el}(W^v, p, R), \kappa) = \lim_{p \rightarrow \infty} \delta^{el}(W^v, p, R) \approx \frac{\sum_{j=1}^4 H(\chi(\mathbf{x}_j))}{4} \quad (3.53)$$

These expressions for the Young's Modulus and density are the same as those employed by the MMC method with ersatz material, meaning that the Generalized Geometry Projection approach could effectively recover it.

To recover the Geometry Projection formulation, one can consider  $p = 2$  and a generic  $R = r$ . For these values the sampling window becomes a circular sampling window, i.e.  $\mathbf{D} \equiv \mathbf{B}_p r$ . Moreover selecting  $W_i = \Upsilon_i$  by the use of equation (3.46) with the same assumption, i.e. the restriction of  $\partial\omega_i$  to be considered as straight (Bell et al. [Bell et al., 2012]; Norato et al. [Norato, 2015]; Zhang et al. [Zhang et al., 2016a]) one can find the expression of the local volume fraction  $\{\delta^{el}\}$ .<sup>6</sup> In order to compute local densities let's set:

$$\rho^{el} = \mathbb{V}(\{\delta\}^{el}, \kappa) = \Pi(\{\hat{\delta}^{el}(r, \gamma_v)\}, \kappa) \quad (3.54)$$

And for the local Young's modulus:

$$E^{el} = \mathbb{M}(\{\delta\}^{el}, E, E_{min}, \kappa) = \Pi(\{\hat{\delta}^{el}(r, \gamma_c)\}, \kappa) E \quad (3.55)$$

Where equation (3.25) is used to compute  $\{\hat{\delta}^{el}\}$ .

Finally, the proposed unified approach can also recover MNA. In fact setting  $W_i = m_i w(v_i, h_i, \varepsilon_i)$ ,  $p \rightarrow \infty$  and  $R = \frac{1}{2}$  and using numerical integration for the integrals in equation (3.46) with just a single Gauss point one gets:

$$\delta_i^{el} \approx W_i = m_i^\gamma w_i^{el} \quad (3.56)$$

This time for the local densities:

$$\mathbb{V}(\{\delta^{el}\}, \kappa) = \Pi(\{\delta^{el}\}_v, \kappa) \quad (3.57)$$

And for the Young's modulus:

$$\mathbb{M}(\{\delta^{el}\}, \kappa) = E_{min} + (E - E_{min}) \Pi(\{\delta^{el}\}_c, \kappa)^{p_b} \quad (3.58)$$

A summary of the parameters to be used in the proposed Generalized Geometry Projection approach to recover all of the three reviewed methods is provided in table 3.1.

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<sup>6</sup>The reader can note that the same result can also be obtained selecting  $p \rightarrow \infty$ , 1 Gauss point,  $R = \frac{1}{2}dx$  and  $W_i^{el} = \delta_i^{el}$  of equation (3.23).

Note that for the proposed GGP approach, it is not only possible to recover existing strate-

Table 3.1: Choice to be made to recover all other approaches using Generalized Geometry Projection

Method	MMC	GP	MNA
$W^c$	$H_\epsilon(\chi^{el})^q$	$\tilde{\delta}_i^{el} m_i^{\gamma_c}$	$m_i^{\gamma_c} w_i^{el}$
$W^v$	$H_\epsilon(\chi^{el})$	$\tilde{\delta}_i^{el} m_i^{\gamma_v}$	$m_i^{\gamma_v} w_i^{el}$
$p$	$\infty$	$\infty$	$\infty$
$R$	$\frac{\sqrt{3}}{2}dx$	$\frac{1}{2}dx$	$\frac{1}{2}dx$
$N_{GP}$	4	1	1
$\mathbb{V}$	$\frac{\sum_{j=1}^4 H_\epsilon(\chi_j^{el})}{4}$	$\Pi(\{\hat{\delta}^{el}\}_v, \kappa)$	$\Pi(\{\delta^{el}\}_v, \kappa)$
$\mathbb{M}$	$\frac{\sum_{j=1}^4 (H_\epsilon(\chi_j^{el}))^q}{4}$	$\Pi(\{\hat{\delta}^{el}\}_c, \kappa)E$	$E_{min} + (E - E_{min})\Pi(\{\delta^{el}\}_c, \kappa)^{pb}$

gies, but it is also possible to adapt an existing technique by changing only  $R$  and  $N_{GP}$ , in order to potentially improve the analysis and optimization behavior. This work will then refer to: Adapted Moving Morphable Components method (AMMC), Adapted Geometry Projection (AGP) and to Adapted Moving Node Approach (AMNA), when using respectively MMC, GP or MNA parameters in table 3.1 with the only exceptions of number of Gauss points in each sampling window  $N_{GP}$  and of the sampling window size  $R$ . In figure 3.9 the Adapted Moving Node Approach is applied to the same example considered in figure 3.1a to compute  $\rho^{el}$  distribution on a uniform  $50 \times 50$  mesh and investigates the variation of both  $R$  and  $N_{GP}$ . In this case MNA characteristic function was considered with a relatively small  $\epsilon$ . When just one Gauss point is employed for the numerical integration,  $R$  has no effect on the final  $\rho^{el}$  distribution. One can observe that increasing  $N_{GP}$  smoothens the  $\rho^{el}$  variations around the bar ends. On the other hand increasing the value of  $R$  smoothens the variation between full and voids elements. These effects are important from the simulation and optimization point of view as will be pointed out in the implementation section.

## 3.6 Geometric assembly

As discussed in the previous subsections, several approaches were reviewed for making the union or the assembly of geometric primitives. It is important to point out, as it was done by Norato et al. [Norato, 2015] , that the assembly of 2D components can be seen as a merging operation (the component's thickness doesn't change at the intersection) or as an overlapping operation (the component thickness is summed) . In the first case one is interested in determining the union of the geometry produced by each component, while in the second case the geometry are simply overlapped in the out of plane direction. Since the stiffness matrix of each finite element is proportional to both the Young's modulus and the out of plane thickness, the local volume fraction of each component could be simply summed up in this case and used to determine an equivalent Young's Modulus that takes into account both the effect of material and out of plane thickness. This ambiguity does not exist in

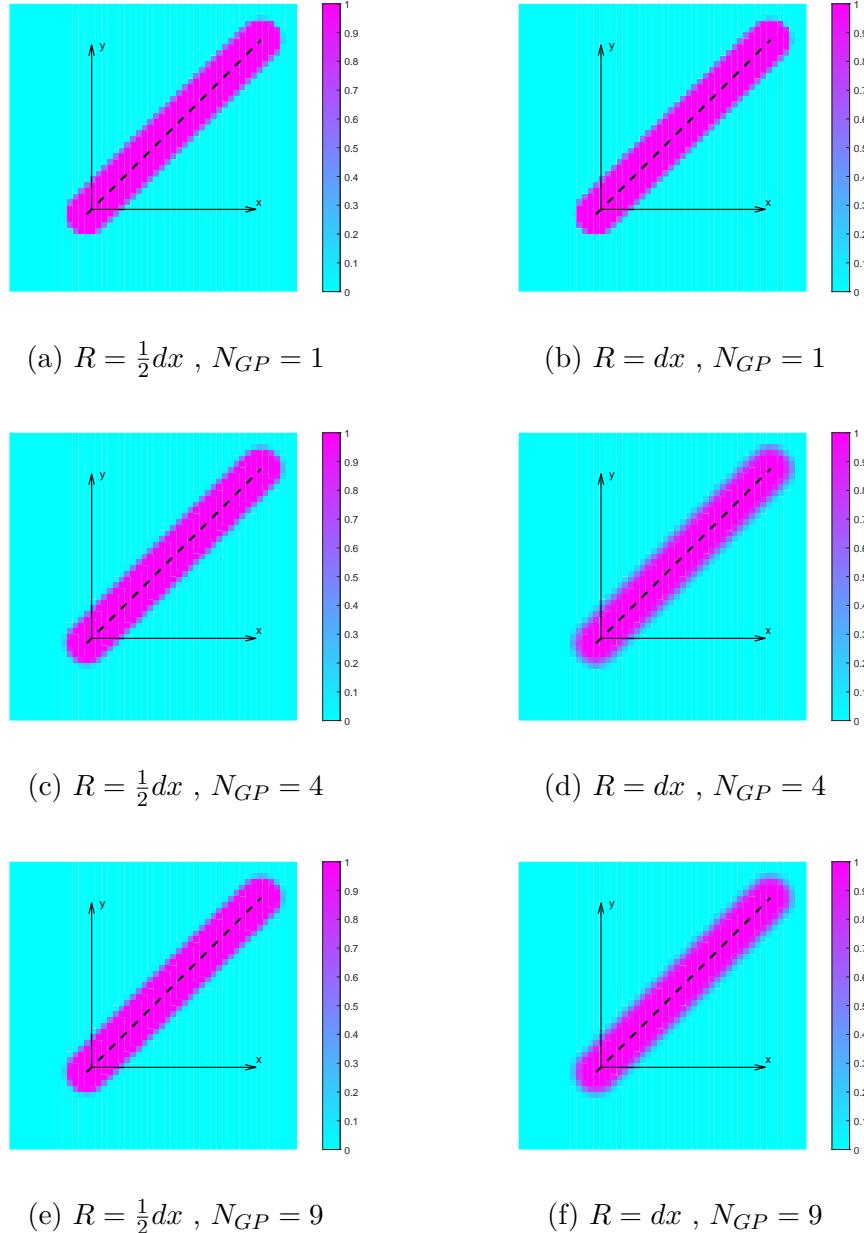


Figure 3.9: Distribution of  $\rho^{el}$  for the generic component of figure 3.1a and for a  $50 \times 50$  mesh over the domain of  $X_g$  for carrying number of Gauss points. MNA characteristic functions were considered with  $X = 1, Y = 1, L = 3, h = 0.5, \theta = \frac{\pi}{4}, \gamma = 3, \varepsilon = 0.07$ . The mesh size  $dx$  along the x direction was considered as  $dx = 0.07$ .

3D topology optimization where the only possibility to make the assembly is the merging strategy. The rest of this paragraph will then focus on the strategy that can be adopted to merge component's geometries. Firstly one can observe that depending on the considered approach, the assembly is carried out either on local volume fractions or more indirectly

through topology description functions (TDF) as in Zhang et al. [Zhang et al., 2016c].<sup>7</sup> In [Zhang et al., 2017g] a comprehensive study of application of Boolean operations is reviewed for implicit geometry description (R-functions or TDF). In this context let's first consider the case when the geometric assembly is applied at the level of the density field. The main advantage of this case consists in being able to treat the local volume fraction coming from each component projection as a pseudo-logical value. In fact let's consider an input vector  $\{z\} \in \{0, 1\}^n$ . In order to make the logic union of all the entry vectors, one can consider either one of the following approaches:

$$\Pi_a(\{z\}) = \min \left( 1, \sum_{i=1}^n z_i \right) \quad (3.59)$$

$$\Pi_b(\{z\}) = 1 - \prod_{i=1}^n (1 - z_i) \quad (3.60)$$

$$\Pi_{max}(\{z\}) = \max_i z_i \quad (3.61)$$

For a logical entry vector  $\{z\} \in \{0, 1\}^n$ ,  $\Pi_a(\{z\}) = \Pi_b(\{z\}) = \Pi_{max}(\{z\})$ . On the other hand when  $\{\mathbf{z}\} \in ]0, 1[^n$  then  $\Pi_a(\{z\}) \neq \Pi_b(\{z\}) \neq \Pi_{max}(\{z\})$ . The asymptotic density operator  $\Pi_a(\{z\})$  first makes an overlap of the component's density and then saturate the result to 1. This approach was employed in the master thesis of Overvelde [Overvelde, 2012]. The minimum between 1 and the value of the sum can be realized by a differentiable saturation function that is detailed later in this paragraph. The Boolean operator  $\Pi_b(\{z\})$  can be recognized as the one employed by the MMB [Hoang and Jang, 2017] approach in order to make the union of bar components. For the third approach, since the maximum is an irregular function, in the literature it is often replaced with its regular approximations. In the context of topology optimization it is possible to cite the p-norm and the p-mean [Duysinx and Sigmund, 1998], the Kreisselmeier–Steinhauser (KS) functional [Kreisselmeier and Steinhauer, 1980] and the more recent induced approaches [Kennedy and Hicken, 2015]. Often these approximations are employed in structural optimization with stress constraints in order to reduce the number of constraints in the optimization problem. Here these methods and some important properties relative to the maximum operator are reviewed. Given an input vector  $\{\mathbf{z}\} \in \mathbb{R}^n$ , and the constant of aggregation  $\kappa \in \mathbb{R}^+$ , the smooth approximation  $\Pi$  of the maximum operator is defined as:  $\Pi : (\mathbb{R}^n, \mathbb{R}^+) \rightarrow \mathbb{R} \mid (\{\mathbf{z}\}, \kappa) \rightarrow \Pi(\{\mathbf{z}\}, \kappa)$  and

$$\lim_{\kappa \rightarrow \infty} \Pi(\{\mathbf{z}\}, \kappa) = \max(\{\mathbf{z}\}) = z_{max} \quad (3.62)$$

Let's consider the p-norm  $\Pi_{pm}$  and the p-mean  $\Pi_{pn}$  [Duysinx and Sigmund, 1998]. For these approaches one can make the assumption that an input vector  $\{\mathbf{z}\}$  has non negative components, thus:

$$\Pi_{pm}(\{\mathbf{z}\}, \kappa) = \left( \frac{1}{n} \sum_{j=1}^n z_j^\kappa \right)^{\frac{1}{\kappa}} \leq z_{max} < \left( \sum_{j=1}^n z_j^\kappa \right)^{\frac{1}{\kappa}} = \Pi_{pn}(\{\mathbf{z}\}, \kappa) \quad (3.63)$$

---

<sup>7</sup>The characteristic function of the union of sets can be easily computed as the maximum of the characteristic functions of each set. The same can be stated for TDFs.

One can also have negative inputs but a double correction has to be made in order to have all non-negative inputs when elevating to the power  $\kappa$ . For instance one can choose a positive value  $z_p$  so that  $z_j + z_p > 0 \forall j = 1, 2, \dots, n$  with this modification one can compute:

$$\Pi_{pm}^p(\{\mathbf{z}\}, \kappa, z_p) = \left( \frac{1}{n} \sum_{j=1}^n (z_j + z_p)^\kappa \right)^{\frac{1}{\kappa}} - z_p \quad (3.64)$$

$$\Pi_{pn}^p(\{\mathbf{z}\}, \kappa, z_p) = \left( \sum_{j=1}^n (z_j + z_p)^\kappa \right)^{\frac{1}{\kappa}} - z_p \quad (3.65)$$

Here both lower bound KS function  $\Pi_{KS}^l$  and the KS function  $\Pi_{KS}$  [Kreisselmeier and Steinhauser, 1980] are also reviewed:

$$\Pi_{KS}^l(\{\mathbf{z}\}, \kappa) = \frac{1}{\kappa} \log \left( \frac{1}{n} \sum_{j=1}^n e^{\kappa z_j} \right) \leq z_{max} < \frac{1}{\kappa} \log \left( \sum_{j=1}^n e^{\kappa z_j} \right) = \Pi_{KS}(\{\mathbf{z}\}, \kappa) \quad (3.66)$$

Finally the induced exponential  $\Pi_{IE}$  [Kennedy and Hicken, 2015] was also considered:

$$\Pi_{IE}(\{\mathbf{z}\}, \kappa) = \frac{\sum_{j=1}^n z_j e^{\kappa z_j}}{\sum_{j=1}^n e^{\kappa z_j}} \leq z_{max} \quad (3.67)$$

In figure 3.10, 3.11, 3.12 and 3.13 all reviewed operators are applied to the vector  $\{\mathbf{z}\} = \{x, 10xe^{1-10x}, 4x(1-x)\}$  for  $x \in [0, 1]$ . A first important remark is that p-mean, lower

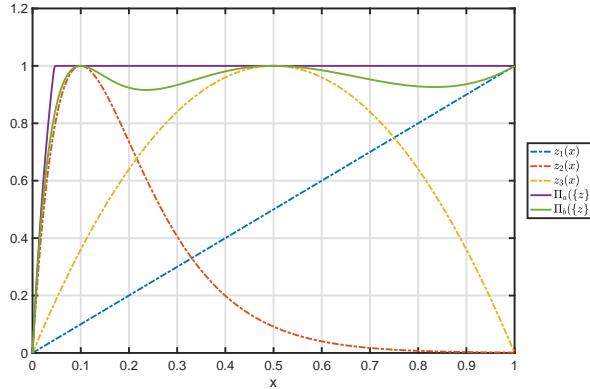


Figure 3.10: Application of asymptotic density operator  $\Pi_a$  and of Boolean operator  $\Pi_b$  to the vector  $\mathbf{z}$  of 3 variables,  $z_1(x) = x$ ,  $z_2(x) = 10xe^{1-10x}$ ,  $z_3(x) = 4x(1-x)$ . The minimum function necessary to carry out the evaluation of  $\Pi_a$  is replaced by the regular approximation of the saturation function of equation (3.71).

bound KS function and induced exponential operator are always less than or equal to the maximum. The equality being true in the case of uniform value for the entry vector i.e.:

$$\Pi_{pm}(\{\mathbf{z}\}, \kappa) = \Pi_{KS}^l(\{\mathbf{z}\}, \kappa) = \Pi_{IE}(\{\mathbf{z}\}, \kappa) = z_{max} \Leftrightarrow z_1 = z_2 = \dots = z_n = z_{max} \quad (3.68)$$

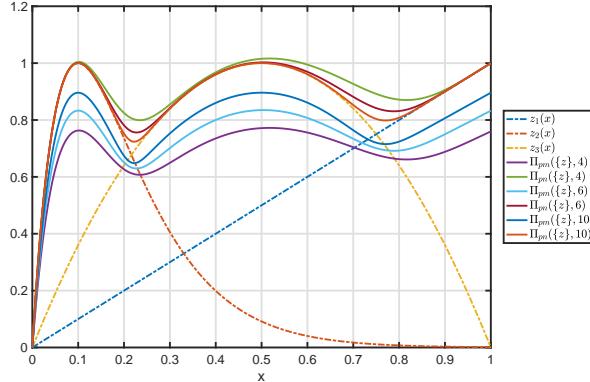


Figure 3.11: Application of p-norm  $\Pi_{pn}$  and p-mean  $\Pi_{pm}$  operator for  $\kappa = 4, 6, 10$  to the vector  $\mathbf{z}$  of 3 variables,  $z_1(x) = x$ ,  $z_2(x) = 10xe^{1-10x}$ ,  $z_3(x) = 4x(1-x)$ .

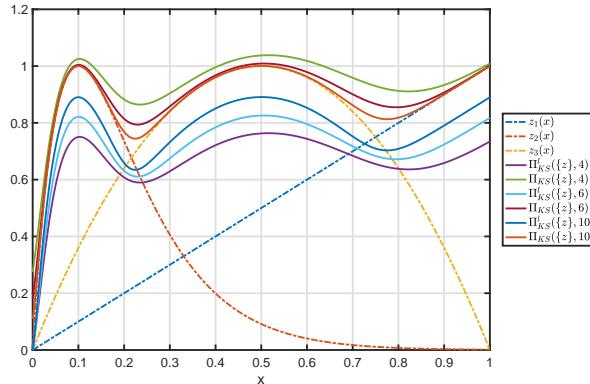


Figure 3.12: Application of KS  $\Pi_{KS}$  and lower bound KS  $\Pi'_{KS}$  function for  $\kappa = 4, 6, 10$  to the vector  $\mathbf{z}$  of 3 variables,  $z_1(x) = x$ ,  $z_2(x) = 10xe^{1-10x}$ ,  $z_3(x) = 4x(1-x)$ .

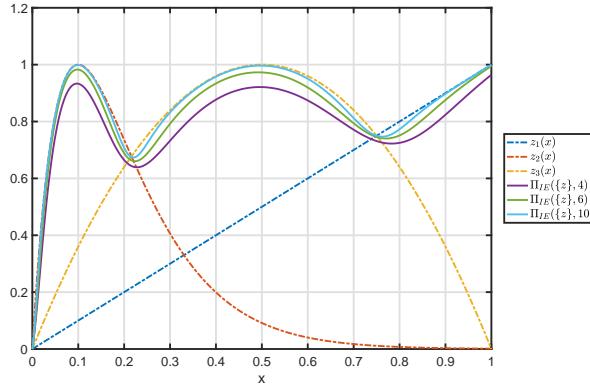


Figure 3.13: Application of induced exponential aggregation function  $\Pi_{IE}$  for  $\kappa = 4, 6, 10$  to the vector  $\mathbf{z}$  of 3 variables,  $z_1(x) = x$ ,  $z_2(x) = 10xe^{1-10x}$ ,  $z_3(x) = 4x(1-x)$ .

On the other hand Boolean, asymptotic density, p-norm and KS functions are always strictly greater than the maximum function. A second remark is that most reviewed approaches (with the exceptions of the asymptotic density and the Boolean operator) produce differentiable

approximations of the maximum function and that by increasing the value of the aggregation constant  $\kappa$ , all approaches tend to recover the maximum function as requested in equation (3.62). At the same time high values of  $\kappa$  reduces the smoothness of the approximated function. A third important remark is that the absolute value of the discrepancy between  $z_{max}$  and the approximation are maximized in the following way depending on the aggregation approach employed:

- For p-norm and KS function the worst accuracy corresponds to a uniform entry vector  $\mathbf{z}$ .
- For p-mean and lower bound KS function the worst accuracy corresponds to a vector  $\mathbf{z}$  so that:  $z_i = u_b \Leftrightarrow i = i_{max}$  and  $z_i = l_b \forall i \neq i_{max}$ . Where  $u_b$  and  $l_b$  are respectively the lower and the upper bound of entry vector components.
- For the induced exponential operator the worst accuracy is reached in intermediate cases

Coming back to the application of these functions to the component assembly, one should consider the following implications:

- Asymptotic density and Boolean operator are thought to be used on pseudo-logical input, i.e.  $\in [0, 1]$ . They could also be adopted for TDF but some re-scaling of the input should be adopted to have a meaningful behavior of both operators.
- If the assembly is made on TDFs as in Zhang et al. [Zhang et al., 2016c], p-norm and p-mean cannot be employed as they are, but they need to be modified in order to avoid a negative value of the argument of the power.
- When considering TDF applications, one is free to use both KS functions and the induced exponential operator without particular numerical difficulties.
- On the other hand when one wants to apply these functions directly to local volume fractions, some difficulties arise. As the final local density must be lower than 1, the KS function and the p-norm cannot be used since their output can possibly be greater than 1. On the other hand p-mean, lower bound KS function and induced exponential can be used, but in the case of completely non overlapped components the resulting maximal projected density can be inferior to 1. This affects the projected Finite Element Model stiffness and depending on the number of components that describe the solution, can produce inaccurate values of the final compliance. Nevertheless one can control this gap increasing the value of  $\kappa$ . On the other hand as aforementioned this plays a role on the projection smoothness, which can increase or prevent optimization convergence.

To overcome these issues, here it is proposed to apply a differentiable approximation of the saturation function to the aggregation operator. This function is defined as:

$$S_t(x) = \min \left( 1, \max \left( \frac{x}{\tilde{x}}, 0 \right) \right) \quad (3.69)$$

As it is non differentiable, it is further proposed to replace it with the KS approximation:

$$s_b(x, \kappa_s) = -\frac{1}{\kappa_s} \log \left( \exp(-\kappa_s) + \frac{1}{1 + \exp(\kappa_s \frac{x}{\tilde{x}})} \right) \quad (3.70)$$

where  $\kappa_s$  is an aggregation constant that can be chosen to be very high ( $\kappa_s \geq 100$ ) in order to get good approximation of  $S_t(x)$ . To improve the model accuracy in case of absence of material, a re-scaling of the saturation function is applied i.e.:

$$s_t(x, \kappa_s) = \frac{s_b(x, \kappa_s) - s_b(0, \kappa_s)}{1 - s_b(0, \kappa_s)} \quad (3.71)$$

This saturation function is represented for several values of the parameter  $\tilde{x}$  in figure 3.14.

Finally the saturated operator will be defined as:

$$P_s(\{z\}, \kappa, \kappa_s) = s_t(\Pi(\{z\}, \kappa), \kappa_s) \quad (3.72)$$

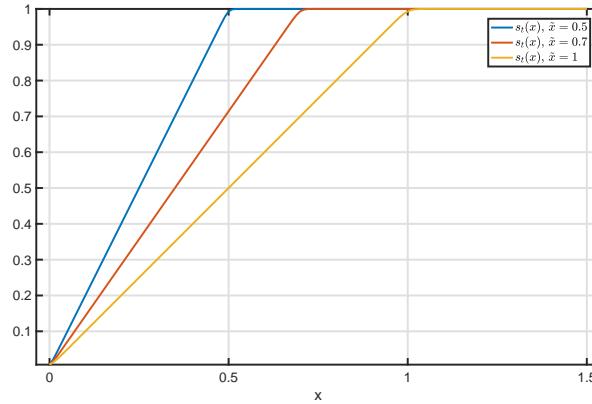


Figure 3.14: Proposed saturation function  $s_t(x)$  for  $\kappa_s = 100$ ,  $\tilde{x} = 0.5$ ,  $\tilde{x} = 0.7$  and  $\tilde{x} = 1$ .

Note that the value  $\tilde{x}$  has to be chosen in order to ensure that, when only one component is projected on the mesh element's centroid, the maximum value of projected local volume fraction is still equal to 1:

$$\tilde{x}_a = \tilde{x}_{KS} = \tilde{x}_{pn} = 1 \quad (3.73)$$

$$\tilde{x}_{KS}^l = 1 + \frac{1}{\kappa} \log \left( \frac{1 + (n-1)e^{-\kappa}}{n} \right) \quad (3.74)$$

$$\tilde{x}_{pm} = \left( \frac{(n-1) z_p^\kappa + (z_p+1)^\kappa}{n} \right)^{\frac{1}{\kappa}} - z_p \quad (3.75)$$

$$\tilde{x}_{IE} = \frac{1}{1 + (n-1)e^{-\kappa}} \quad (3.76)$$

The results of the application of the saturation function is applied to the smooth approxima-

tion of the maximum operator and is illustrated in figures 3.15,3.16,3.17 . One can observe

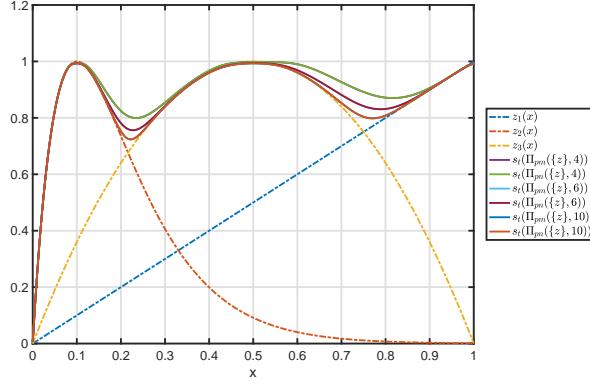


Figure 3.15: Application of the proposed saturated p-norm  $s_t(\Pi_{pn})$  and p-mean  $s_t(\Pi_{pm})$  operator for  $\kappa = 4, 6, 10$  to the vector  $\mathbf{z}$  of 3 variables,  $z_1(x) = x, z_2(x) = 10xe^{1-10x}, z_3(x) = 4x(1-x)$ . One can observe the effect of the saturation since, there are no saturated value greater than 1. Moreover for  $x = 1$  and  $x = 0$ , when just one component of the vector  $\{\mathbf{z}\}$  are equal to 1 and the others are 0, the saturation ensures again a value of 1.

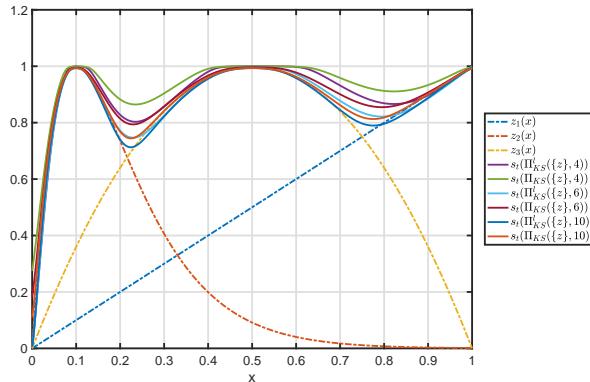


Figure 3.16: Application of the proposed saturated KS  $s_t(\Pi_{KS})$  and lower bound KS  $s_t(\Pi_{KS}^l)$  function for  $\kappa = 4, 6, 10$  to the vector  $\{\mathbf{z}\}$  of 3 variables,  $z_1(x) = x, z_2(x) = 10xe^{1-10x}, z_3(x) = 4x(1-x)$ . One can observe the effect of the saturation since, there are no saturated value greater than 1. Moreover for  $x = 1$  and  $x = 0$ , when just one component of the vector  $\{\mathbf{z}\}$  are equal to 1 and the others are 0, the saturation ensures again a value of 1.

that even when only one component projects to a local volume fraction of 1, the corresponding saturated local volume fraction is nearly at the value of 1 for all reviewed approaches. In order to show the effect of the components assembly and of the saturation, several approaches have been tested on the same configuration and the results presented in figure 3.18. Another important remark is that without the saturation the p-mean, lower bound KS and induced exponential operator do not achieve the desired final density, i.e.  $\rho^{el}$  is not 1 when just one of the two components projects to a density of 1. On the other hand this issue is correctly addressed by the saturation procedure proposed here.

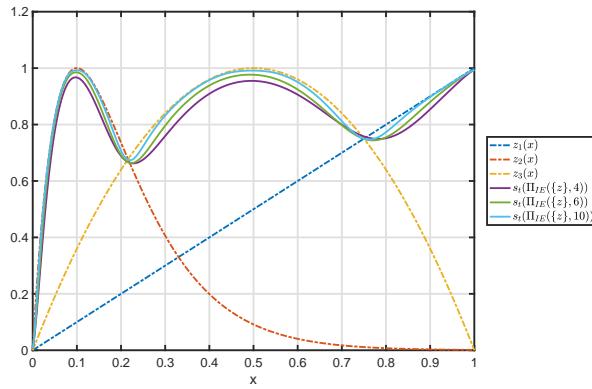


Figure 3.17: Application of the proposed saturated induced exponential aggregation function  $s_t(\Pi_{IE})$  for  $\kappa = 4, 6, 10$  to the vector  $\mathbf{z}$  of 3 variables,  $z_1(x) = x$ ,  $z_2(x) = 10xe^{1-10x}$ ,  $z_3(x) = 4x(1-x)$ . One can observe the effect of the saturation since, there are no saturated values greater than 1. Moreover for  $x = 1$  and  $x = 0$ , when just one component of the vector  $\{z\}$  are equal to 1 and the others are 0, the saturation ensures again a value of 1.

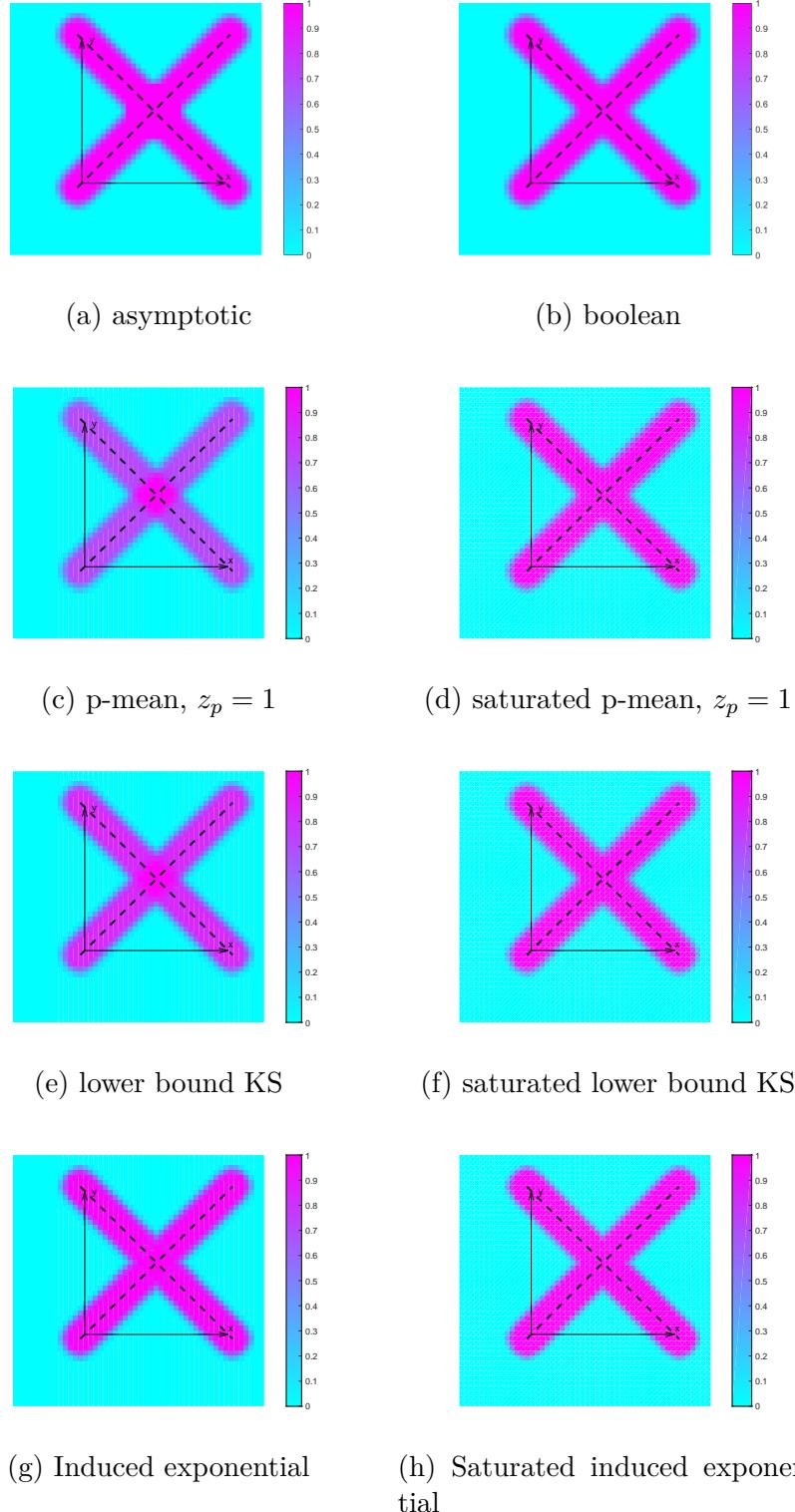


Figure 3.18: Distribution of  $\rho^{el}$  for two components for various assembly operators and application or not of saturation. One of the components is in the same configuration of figure 3.1a, the other with  $\theta = -\frac{\pi}{4}$  and for a  $50 \times 50$  mesh over the domain of  $X_g$ . AMNA with  $, \gamma = 3, \varepsilon = 0.07, \kappa = 4, \kappa_s = 100$  was considered. Where  $dx = 0.07$  is the mesh size along the x direction.

### 3.7 Sensitivity analysis

This subsection derives and analyzes the gradient of the responses (compliance  $C$  and volume fraction  $V$ ) involved in solving the topology optimization problem considered in equation (3.13). The aim of this sensitivity analysis is to provide recommendations with respect to good parameter choices in the proposed Generalized Geometry Projection method.

Compliance  $C$  and volume fraction  $V$  sensitivities are compute by chain rule:

$$\frac{\partial C}{\partial x_i} = \sum_{el=1}^{N_{el}} \frac{\partial C}{\partial E^{el}} \frac{\partial E^{el}}{\partial x_i} \quad (3.77)$$

$$\frac{\partial V}{\partial x_i} = \sum_{el=1}^{N_{el}} \frac{\partial V}{\partial \rho^{el}} \frac{\partial \rho^{el}}{\partial x_i} \quad (3.78)$$

where  $N_{el}$  is the number of finite element in the mesh. This can compactly be reformulated as:

$$\left\{ \frac{\partial C}{\partial x} \right\} = \left[ \frac{\partial E}{\partial x} \right] \left\{ \frac{\partial C}{\partial E} \right\} \quad (3.79)$$

$$\left\{ \frac{\partial V}{\partial x} \right\} = \left[ \frac{\partial \rho}{\partial x} \right] \left\{ \frac{\partial V}{\partial \rho} \right\} \quad (3.80)$$

In equations (3.79),(3.80), the right hand side vectors (of size  $N_{el} \times 1$ ) are not different from the one computed for density based topology optimization. On the other hand the matrices  $\left[ \frac{\partial E}{\partial x} \right]$ ,  $\left[ \frac{\partial \rho}{\partial x} \right]$  ( of size  $N_v \times N_{el}$ ) are specific to the Generalized Geometry Projection method. Let's derive their analytic expression:

$$\left[ \frac{\partial E}{\partial x} \right] = \left[ \frac{\partial \mathbb{M}}{\partial x} \right] = \sum_{i=1}^n \left[ \frac{\partial \delta_i}{\partial x} \right] \left[ \frac{\partial \mathbb{M}}{\partial \delta_i} \right] \quad (3.81)$$

$$\left[ \frac{\partial \rho}{\partial x} \right] = \left[ \frac{\partial \mathbb{V}}{\partial x} \right] = \sum_{i=1}^n \left[ \frac{\partial \delta_i}{\partial x} \right] \left[ \frac{\partial \mathbb{V}}{\partial \delta_i} \right] \quad (3.82)$$

where  $\left[ \frac{\partial \mathbb{M}}{\partial \delta_i} \right]$  and  $\left[ \frac{\partial \mathbb{V}}{\partial \delta_i} \right]$  are  $N_{el} \times N_{el}$  diagonal matrices and the terms  $\left[ \frac{\partial \delta_i}{\partial x} \right]$  are matrices of size  $N_v \times N_{el}$ . A first important observation is that  $\left[ \frac{\partial \delta_i}{\partial x} \right]$  is sparse and only the lines of variables belonging to the  $i^{th}$  component will be different from zero i.e.  $\left[ \frac{\partial \delta_i}{\partial x_i} \right]$  ( $6 \times N_{el}$ ). This means that each row of  $\left[ \frac{\partial E}{\partial x} \right]$ ,  $\left[ \frac{\partial \rho}{\partial x} \right]$  will have just one contribution coming from the component defined by the corresponding variable. Let's first derive the terms in the diagonal of  $\left[ \frac{\partial \mathbb{M}}{\partial \delta_i} \right]$  and  $\left[ \frac{\partial \mathbb{V}}{\partial \delta_i} \right]$ . As an example MNA characteristic functions with the saturation function applied after the geometry assembly are considered:

$$\frac{\partial \mathbb{M}^{el}}{\partial \delta_i^{el}} = \frac{\partial \mathbb{M}^{el}}{\partial P_s} \frac{\partial P_s}{\partial \Pi} \frac{\partial \Pi}{\partial \delta_i^{el}} \quad (3.83)$$

$$\frac{\partial \mathbb{V}^{el}}{\partial \delta_i^{el}} = \frac{\partial \mathbb{V}^{el}}{\partial P_s} \frac{\partial P_s}{\partial \Pi} \frac{\partial \Pi}{\partial \delta_i^{el}} \quad (3.84)$$

The evaluation of  $\frac{\partial \mathbb{M}^{el}}{\partial s_t}$  and of  $\frac{\partial \mathbb{V}^{el}}{\partial s_t}$  depends on the choice made among the existing functions. For AMNA one gets:

$$\frac{\partial \mathbb{M}^{el}}{\partial P_s} = p_b(E - E_{min}) s_t^{p_b-1} \quad (3.85)$$

$$\frac{\partial \mathbb{V}^{el}}{\partial P_s} = 1 \quad (3.86)$$

For the saturation function one can get:

$$\frac{\partial P_s}{\partial \Pi} = \frac{\exp\left(\kappa_s \frac{\Pi}{\tilde{\Pi}}\right) \left(\exp\left(\kappa_s \frac{\Pi}{\tilde{\Pi}}\right) + 1\right)^{-2}}{\tilde{\Pi} \left(\exp(-\kappa_s) + \frac{1}{\exp(\kappa_s \frac{\Pi}{\tilde{\Pi}}) + 1}\right)} \frac{1}{1 - s_b(0, \kappa_s)} \quad (3.87)$$

Where  $\tilde{\Pi}$  has to be computed according to equations (3.73)-(3.76). For the computation of  $\frac{\partial \Pi}{\partial \delta_i^{el}}$  the interested reader can find the computation for  $KS_l$ ,  $KS$  and induced exponential in [Kennedy and Hicken, 2015]. Details are provided for p-norm and p-mean computations as follows:

$$\frac{\partial \Pi_{pm}^p}{\partial \delta_i^{el}} = \frac{1}{n} \left( \delta_i^{el} + z_p \right)^{\kappa-1} \left( \frac{1}{n} \sum_{j=1}^n \left( \delta_j^{el} + z_p \right)^\kappa \right)^{\frac{1}{\kappa}-1} \quad (3.88)$$

$$\frac{\partial \Pi_{pn}^p}{\partial \delta_i^{el}} = \left( \delta_i^{el} + z_p \right)^{\kappa-1} \left( \sum_{j=1}^n \left( \delta_j^{el} + z_p \right)^\kappa \right)^{\frac{1}{\kappa}-1} \quad (3.89)$$

For the term  $\left[ \frac{\partial \delta_i}{\partial x_i} \right]$  when using Gauss quadrature one has:

$$\left[ \frac{\partial \delta_i}{\partial x_i} \right] = \frac{\sum_{k=1}^{N_{GP}} \psi_k \left[ \frac{\partial W_{ik}}{\partial x_i} \right]}{\sum_{k=1}^{N_{GP}} \psi_k} \quad (3.90)$$

Finally for the derivatives of  $\left[ \frac{\partial W_{ik}}{\partial x_i} \right]$  one can again use the chain rule and the hypothesis of MNA characteristic functions. For each row of  $\left[ \frac{\partial W_{ik}}{\partial x_i} \right]$  one has a different expression depending on which variable is being considered. Note that from now on, index and parenthesis notations are neglected for brevity. It is then possible to derive the analytic expression of each derivatives, knowing that each expression has to be applied to each couple of components and point of Gauss of each sampling window. For  $m$  variables one has in fact:

$$\frac{\partial W}{\partial m} = \gamma m^{\gamma-1} w \quad (3.91)$$

For variable  $h$  using the chain rule:

$$\frac{\partial W}{\partial h} = m^\gamma \left\{ \frac{\partial w}{\partial h} \right\} \quad (3.92)$$

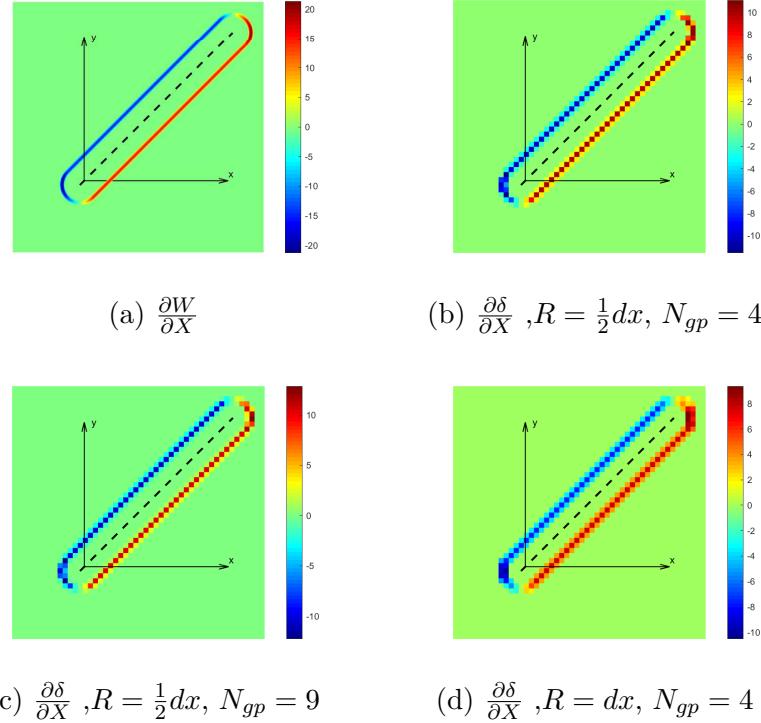


Figure 3.19: Derivatives distribution of  $W$  and  $\delta$  with respect to  $X$  for varying number of Gauss Points  $N_{GP}$  in the sampling window and varying sampling window size  $R$ . It was considered Adapted Moving Node Approach with the generic component of figure 3.1a in the configuration  $X = 1, Y = 1, L = 3, h = 0.5, \theta = \frac{\pi}{4}, \gamma = 3, \varepsilon = 0.07$  and  $dx = 0.07$  for a  $50 \times 50$  mesh over the domain of  $X_g$ .

where

$$\frac{\partial w}{\partial h} = \begin{cases} \frac{\partial a_2}{\partial h} v^2 + \frac{\partial a_1}{\partial h} v + \frac{\partial a_0}{\partial h} & \text{if } l < v < u , \\ 0 & \text{otherwise.} \end{cases} \quad (3.93)$$

and where

$$\frac{\partial a_2}{\partial h} = -\frac{3}{\varepsilon^3} \quad (3.94)$$

$$\frac{\partial a_1}{\partial h} = 6 \frac{h}{\varepsilon^3} \quad (3.95)$$

$$\frac{\partial a_0}{\partial h} = 3 \frac{\varepsilon^2 - h^2}{4\varepsilon^3} \quad (3.96)$$

For the other components, following derivatives are obtained by the chain rule:

$$\frac{\partial W}{\partial X} = m^\gamma \frac{\partial w}{\partial v} \left( \frac{\partial v}{\partial \varrho} \frac{\partial \varrho}{\partial X} + \frac{\partial v}{\partial \phi} \frac{\partial \phi}{\partial X} \right) \quad (3.97)$$

$$\frac{\partial W}{\partial Y} = m_i^\gamma \frac{\partial w}{\partial v} \left( \frac{\partial v}{\partial \varrho} \frac{\partial \varrho}{\partial Y} + \frac{\partial v}{\partial \phi} \frac{\partial \phi}{\partial Y} \right) \quad (3.98)$$

$$\frac{\partial W}{\partial L} = m_i^\gamma \frac{\partial w}{\partial v} \frac{\partial v}{\partial L} \quad (3.99)$$

$$\frac{\partial W}{\partial \theta} = m_i^\gamma \frac{\partial w}{\partial v} \frac{\partial v}{\partial \phi} \frac{\partial \phi}{\partial \theta} \quad (3.100)$$

That can be evaluated by the use of:

$$\frac{\partial w}{\partial v} = \begin{cases} 3a_3v^2 + 2a_2v + a_1 & \text{if } l < v < u , \\ 0 & \text{otherwise.} \end{cases} \quad (3.101)$$

$$\frac{\partial v}{\partial \varrho} = \begin{cases} \frac{2\varrho - L|\cos(\phi)|}{2v} & \text{if } \frac{L^2}{4} < \varrho^2 \cos(\phi)^2 \\ |\sin(\phi)| & \text{if } \frac{L^2}{4} \geq \varrho^2 \cos(\phi)^2 \end{cases} \quad (3.102)$$

$$\frac{\partial v}{\partial \phi} = \begin{cases} \frac{L\varrho \operatorname{sign}(\cos(\phi)) \sin(\phi)}{2v} & \text{if } \frac{L^2}{4} < \varrho^2 \cos(\phi)^2 \\ \varrho \operatorname{sign}(\sin(\phi)) \cos(\phi) & \text{if } \frac{L^2}{4} \geq \varrho^2 \cos(\phi)^2 \end{cases} \quad (3.103)$$

$$\frac{\partial v}{\partial L} = \begin{cases} \frac{L - \varrho|\cos(\phi)|}{2v} & \text{if } \frac{L^2}{4} < \varrho^2 \cos(\phi)^2 \\ 0 & \text{if } \frac{L^2}{4} \geq \varrho^2 \cos(\phi)^2 \end{cases} \quad (3.104)$$

$$\frac{\partial \varrho}{\partial X} = \frac{X - x}{\varrho} \quad (3.105)$$

$$\frac{\partial \varrho}{\partial Y} = \frac{Y - y}{\varrho} \quad (3.106)$$

$$\frac{\partial \phi}{\partial X} = \frac{y - Y}{\varrho^2} \quad (3.107)$$

$$\frac{\partial \phi}{\partial Y} = \frac{X - x}{\varrho^2} \quad (3.108)$$

$$\frac{\partial \phi}{\partial \theta} = -1 \quad (3.109)$$

Let us note that based on these definitions some sensitivities could be either not defined or not continuous. However note that by respecting the condition  $h > \varepsilon$  these issues are avoided. In figures 3.19,C.1,C.2,C.3 and C.4 the Generalized Geometry Projection is employed to study the distribution of gradients of both  $W$  and  $\delta$ . The effect of both sampling window size and number of Gauss points is investigated. It is then possible to make some observations and recommendations:

- All represented gradient components even if defined piece-wisely are differentiable.
- All gradient components of  $W$  take values different from zero only in the component's

transition zone.

- After applying the Generalized Geometry Projection gradient components of  $\delta$  are averaged on the sampling windows and as a consequence are much smaller. Increasing the number of Gauss points, derivatives of  $\delta$  become smoother, which is beneficial for the optimization. Increasing the sampling window size increases the thickness of the transition zone as well as the gradients of  $\delta$ , which also has beneficial effects as will be further illustrated in the next section.

## 3.8 Numerical investigations on 2D use cases

This section investigates, on several numerical applications, the effects of the various parameters (such as number of Gauss points  $N_{GP}$  and sampling windows size  $R$ ) presented in the Generalized Geometry Projection, in terms of finite element analysis accuracy and topology optimization problem ill conditioning. The first subsection considers a simple cantilever beam that can be modelled using both the Geometry projection scheme and classic Euler beam finite element. The aims of this analysis is to investigate the model accuracy and limits when using Generalized Geometry Projection. The topology optimization subsection investigates the behavior of Generalized Geometry Projection to be used for the resolution of a 2D topology optimization problem: the short cantilever beam. This problem has been widely studied by several works, here it is considered only to assess a common problem that every approach is faced with and for which practical recommendations are provided.

### 3.8.1 Parametric study of a cantilever beam

Let's first consider a simple test case that can also be compared to theoretical results: the cantilever beam (c.f. figure 3.20). In table 3.2 the values chosen for numerical experiments are detailed. Here it will be investigated the effect of thickness  $h$ , of the number of Gauss

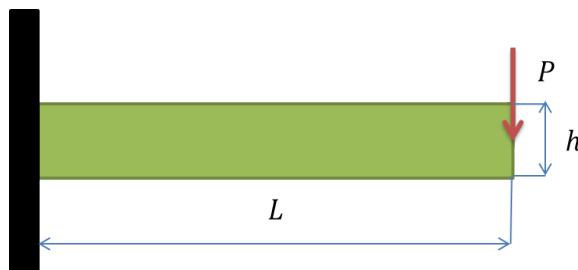


Figure 3.20: Representation of the considered cantilever beam problem.

points  $N_{GP}$  and of the sampling window size (of the Generalized Geometry Projection)  $R$  as well as the effect of the topology optimization method employed (AMMC, AGP or AMNA) on the total compliance  $C$  and volume fraction  $V$  (defined with respect to the total volume occupied by the solid finite element mesh). Using the well-known Euler beam model, one can

Table 3.2: Parameters retained for the parametric study of the cantilever beam

Parameter name	symbol	value
Material Young Modulus	$E$	1
Beam Length	$L$	100
Beam width (out of plane direction)	$b$	1
Beam height	$h$	$\in [1, 10]$
Load amplitude	$P$	1
Number of element in x direction	$n_{elx}$	100
Number of element in y direction	$n_{ely}$	50
Poisson ratio	$\nu$	0.3
element size in x direction	$dx$	1
element size in y direction	$dy$	1
AMMC parameter $\alpha$	$\alpha$	1
AMMC parameter $\beta$	$\beta$	$10^{-3}$
AMMC parameter $\epsilon$	$\epsilon$	0.7
AMMC parameter $q$	$q$	3
AMMC parameter $\alpha$	$\alpha$	1
AGP parameter $r$	$r$	1.5
AGP/AMNA parameter $\gamma_v$	$\gamma_v$	1
AGP/AMNA parameter $\gamma_c$	$\gamma_c$	1.5
AGP parameter $\delta_{min}$	$\delta_{min}$	$10^{-6}$
AMNA parameter $\varepsilon$	$\varepsilon$	3
AMNA parameter $p_b$	$p_b$	1
AMNA parameter $E_{min}$	$E_{min}$	$10^{-6}$
Aggregation constant for saturation	$\kappa_s$	$10^2$

in fact compute analytically the compliance and the volume fraction of the beam:

$$C = \frac{4PL^3}{Ebh^3} = \frac{4 \times 10^6}{h^3} \quad (3.110)$$

$$V = \frac{bLh}{n_{elx}n_{ely}} = \frac{h}{50} \quad (3.111)$$

The reader can observe that for the range of values selected for  $h \in [1, 10]$  the ratio between the beam length  $L$  and its cross section area  $A = bh$  is  $\frac{L}{A} \in [10, 100]$ , large enough to consider the hypothesis of Euler beam model reasonable. For this cantilever beam problem the Generalized Geometry Projection is considered, over the mesh illustrated in figure 3.21b.

The results of this trade study, which varied parameters  $h$ ,  $N_{GP}$ ,  $R$  and the chosen method, are shown in figures 3.22 and appendix figures C.5,C.6 and C.7. It is possible to make the following observations:

- For all methods one gets better accuracy for higher thicknesses, especially for the compliance  $C$ . These effects are closely related to a very well-known issue of solid elements with complete integration: the shear-locking effect. The stiffness of thin structures are

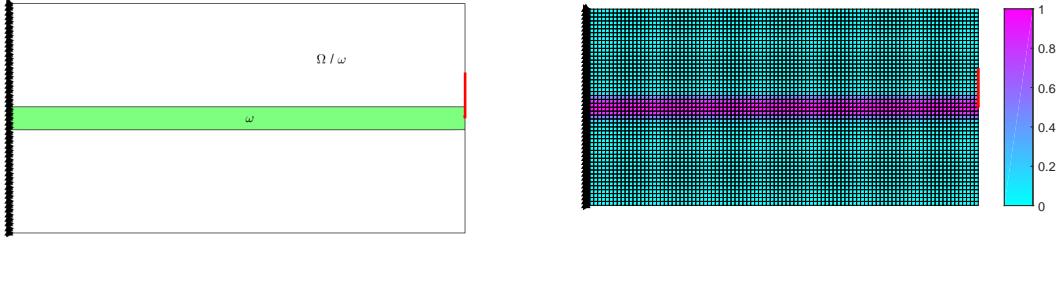


Figure 3.21: Illustration of the Adapted Moving Node Approach (AMNA) for the cantilever beam. A single round ended component  $\omega$  is considered in the configuration  $\{x\} = \{50, 25, 100, 5, 0\}$ . The  $100 \times 50$  2D planar stress solid element mesh covers the domain  $\Omega \cong [0, 100] \times [0, 50]$ .  $N_{GP} = 1$  and  $R = \frac{1}{2}dx$  and the saturation function were employed. The other hyper-parameters are summarized in table 3.2. The round ends of the components fall outside  $\Omega$  and are not represented in these figures.

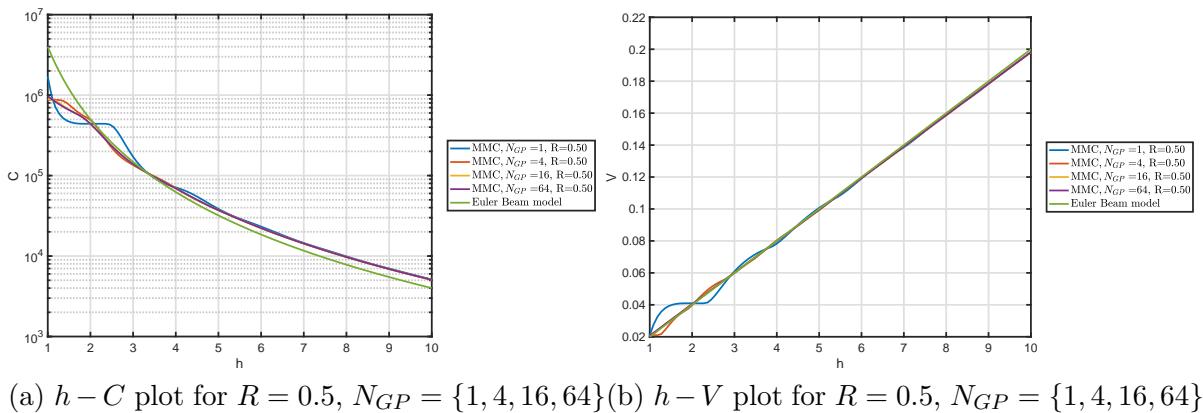


Figure 3.22: Cantilever beam parametric study using the AMMC approach for  $R = 0.5$ . Effect of the sampling window number of Gauss points  $N_{GP}$  on the structural compliance and the volume fraction. In each graph it is reported in green the true theoretical values based on the analytic beam model. The remainder of results can be found in annexes.

overestimated when using few element in the thickness direction. As a consequence the mesh size can control the minimal dimension of the components that one can consider in topology optimization without losing model accuracy.

- For all reviewed adapted methods, mesh induced inflection points in both compliance and volume fraction graph may be observed (i.e. waviness in the respective curves).

This is especially the case for small values of  $h$  and of  $N_{GP}$  and for AMMC.

- Increasing the sampling window size  $R$  (for  $N_{GP} > 1$ ) reduces the model compliance.
- Increasing the number of Gauss points  $N_{GP}$ , mesh induced phenomena are attenuated for all approaches.

The model behavior is dictated by the transition region at the border of the components, especially for small  $h$ , which also explains why AMMC amplifies the mesh induced phenomena. In fact, the transition region thickness being proportional to  $h$ , for small value of  $h$  and  $N_{GP}$ , the transition region is small enough to be located between Gauss point locations. In this situation for small changes of the thickness  $h$  both  $C$  and  $V$  will not change, as can be seen in figure C.5 These observation are of course relative to the particular choice of settings for each approach and are not necessarily generalizable to other geometric configurations. Nevertheless, based on the causes mentioned for these effects, they are likely to reoccur in many other situations. As shown, a good choice of the topology optimization parameters  $h$ ,  $N_{GP}$ ,  $R$  and formulation can however reduce these negative effects. Finally it is possible to point out the fact that these conclusions are consistent with the observation made in the work of Zhou et al. [Zhou et al., 2016] for geometric feature based topology optimization using level set topology optimization.

### 3.8.2 Topology Optimization of the short cantilever beam

This subsection considers the topology optimization of a short cantilever beam, starting from an initial component configuration shown in figure 3.23. The effects of the number of Gauss points in each sampling window  $N_{GP}$  on the optimization convergence will be investigated on this test case, when using the adapted methods based on the proposed GGP framework: AMMC, AGP and AMNA. The well known Method of Moving Asymptotes [Svanberg, 1987] was implemented using the Matlab version distributed by the author [Svanberg, 2004]. A special rescaling was also adopted to avoid MMA numerical instabilities as explained in appendix C.3. A saturated KSI aggregation operator was employed to make the geometric union of component for all adapted methods. The design variables are initiated according to figure 3.23. The stopping criteria employed was on the infinite norm of the configuration variation. The numerical values chosen for all the parameters of both projection and optimization solver are detailed in table 3.3. In figure 3.24, 3.25 and 3.26 the results of the topology optimization problem of a short cantilever beam are considered for several value of  $N_{GP}$ .

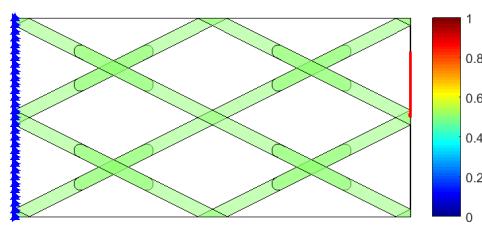


Figure 3.23: Initial configuration for the short cantilever topology optimization problem. Components are colored according to the value of  $m$ . Blue triangles represents clamped degrees of freedoms. The red arrow represents the applied load. 18 round ended bars are considered for the optimization, i.e.  $6 \times 18 = 108$  design variables for both AGP and AMNA and  $5 \times 18 = 90$  design variables for AMMC.

Table 3.3: Parameters used for the parametric study of the short cantilever beam

Parameter name	symbol	value
Material Young Modulus	$E$	1
Design zone width (out of plane direction)	$b$	1
Load amplitude	$P$	1
Number of element in x direction	$n_{elx}$	60
Number of element in y direction	$n_{ely}$	30
Poisson ratio	$\nu$	0.3
element size in x direction	$dx$	1
element size in y direction	$dy$	1
AMMC parameter $\alpha$	$\alpha$	1
AMMC parameter $\beta$	$\beta$	$10^{-3}$
AMMC parameter $\epsilon$	$\epsilon$	0.866
AMMC parameter $q$	$q$	2
AMMC parameter $\alpha$	$\alpha$	1
AGP parameter $r$	$r$	0.5
AGP/AMNA parameter $\gamma_v$	$\gamma_v$	1
AGP/AMNA parameter $\gamma_c$	$\gamma_c$	3
AGP parameter $\delta_{min}$	$\delta_{min}$	$10^{-6}$
AMNA parameter $\varepsilon$	$\varepsilon$	1
AMNA parameter $p_b$	$p_b$	3
AMNA parameter $E_{min}$	$E_{min}$	$10^{-6}$
Aggregation constant for saturation	$\kappa_s$	$10^2$
Aggregation constant	$\kappa$	10
MMA moving limit		0.1
MMA initial moving limit		0.01
MMA incremental factor asyincr		1.2
MMA decremental factor asydecr		0.4
MMA parameter albefa		0.1
MMA parameter move		0.5
Stopping criterion, design variable variation		0.001
Minimal x position	$x_{min}$	-1
Minimal y position	$y_{min}$	-1
Minimal length	$L_{min}$	0
Minimal height	$h_{min}$	1
Minimal angle	$\theta_{min}$	$-2\pi$
Minimal component density	$m_{min}$	0
Maximal x position	$x_{max}$	$n_{elx} + 1$
Maximal y position	$y_{max}$	$n_{ely} + 1$
Maximal length	$L_{max}$	$\sqrt{n_{elx}^2 + n_{ely}^2}$
Maximal height	$h_{max}$	$\sqrt{n_{elx}^2 + n_{ely}^2}$
Maximal angle	$\theta_{max}$	$2\pi$
Maximal component density	$m_{max}$	1

It is well known that many objective functions in topology optimization problems are multi-modal, i.e. there are many local minima. Accordingly one can have convergence to several, sometimes different local optima, depending, here, on both method employed for projection and on the number of Gauss points. In all cases it is possible to observe a convergence to a reasonable local minimum that is consistent with boundary conditions and load application. It is also possible to observe that as a general tendency, by increasing the number of Gauss points one reduces the iterations at convergence and/ or improves the final compliance of the solution. One should keep in mind that the material update law is different for each approach and this may explain why the same geometric configuration can have different values of compliance and volume fraction for each approach. These results confirm the observations already made for the cantilever parametric study. Considering small values of the transition width and small values for  $N_{GP}$ , the optimization solver needs more iterations to converge. Intuitively one can say that this is also a consequence of compliance and volume fraction behaviors that is in these cases perturbed by mesh induced effects.

### 3.8.3 Discussion

The previous results allowed us to investigate the effect of various parameters of the Generalized Geometry Projection (GGP) approach on both simulation and optimization. One of the main points these results highlight is that, depending on the geometric characteristic functions employed (transition region width), and on the number of Gauss points in the sampling window, several mesh induced phenomena can detrimentally impact the model responses. One can use several strategies that aim at reducing these phenomena:

- **Increase the component transition region width** to a size large enough to ensure that at least one Gauss point falls inside it. This strategy is possible for all reviewed approaches, is quite simple, and does not represent any significant numerical expense in terms of neither simulation nor optimization. The main drawback of this strategy is that one needs to consider bigger components for a given mesh size. In fact considering component with a thickness  $h$  smaller than the transition width, does not ensure sensitivity smoothness (For AGP and AMNA). For AMMC the transition width is directly proportional to the component size so that for too small values of  $h$  there is not a value of  $\epsilon$  and  $\alpha$  that can be chosen to give sufficiently high values of the transition width.
- **Refine the mesh** keeping the same value of the transition thickness and of the minimal component size. This is an alternative to the previous recommendation which can be seen as strictly equivalent. In this way, obviously one increases the chance of having a sampling window Gauss point inside transition region, but this comes at a higher computational cost in terms of both memory and CPU time.
- **Use a multi-resolution approach [Liu et al., 2018a]**. These approaches have also the virtue of filling the transition region with Gauss points. This time the difference with GGP is that the assembly of the stiffness matrix is realized using the contribution given by the point inside each element. This means that from a computational burden point

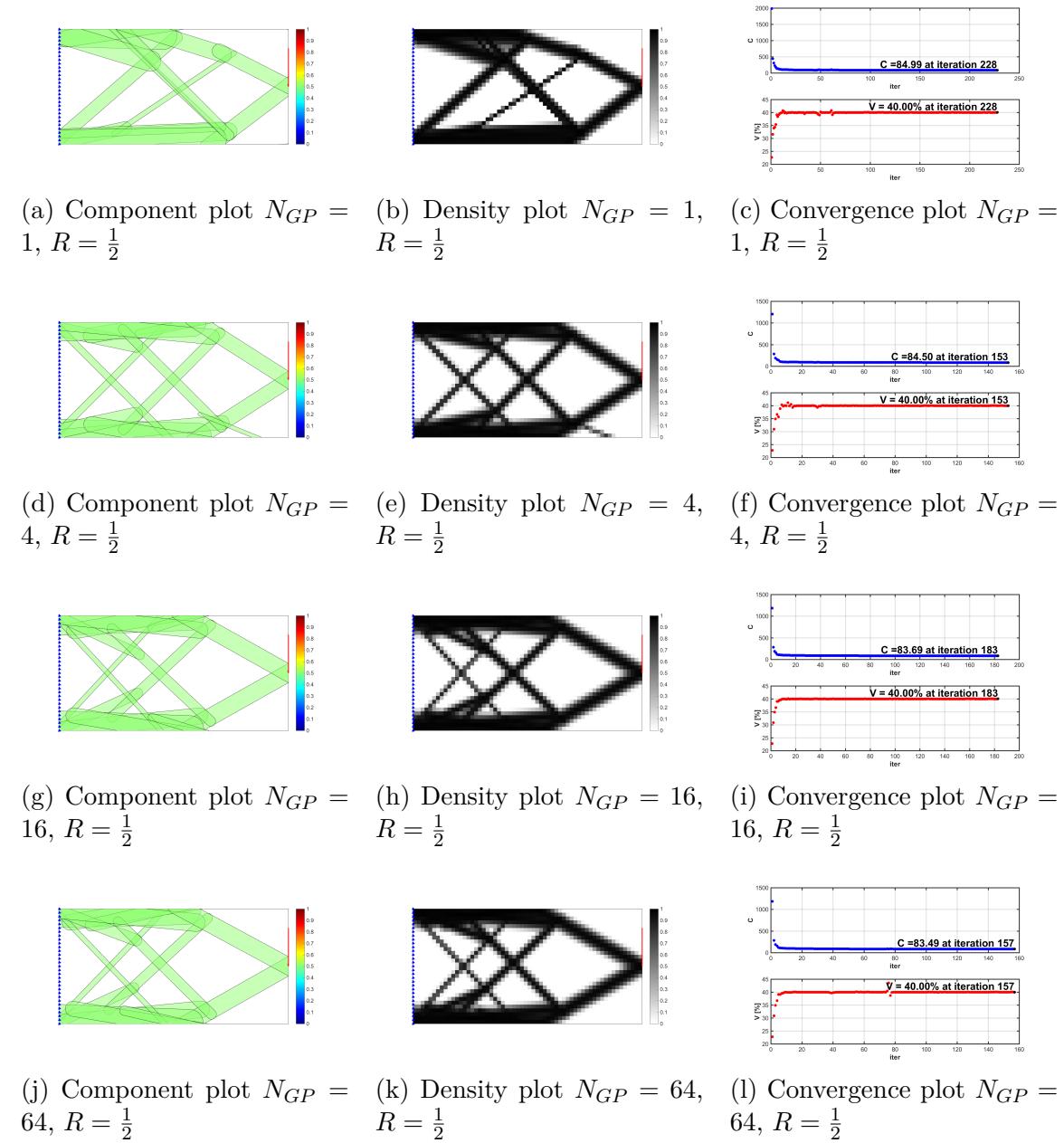


Figure 3.24: Short Cantilever Beam Topology optimization using the AMMC method for variable number of Gauss points  $N_{GP}$ . It must be noted that for AMMC the design is not affected by  $m$ .

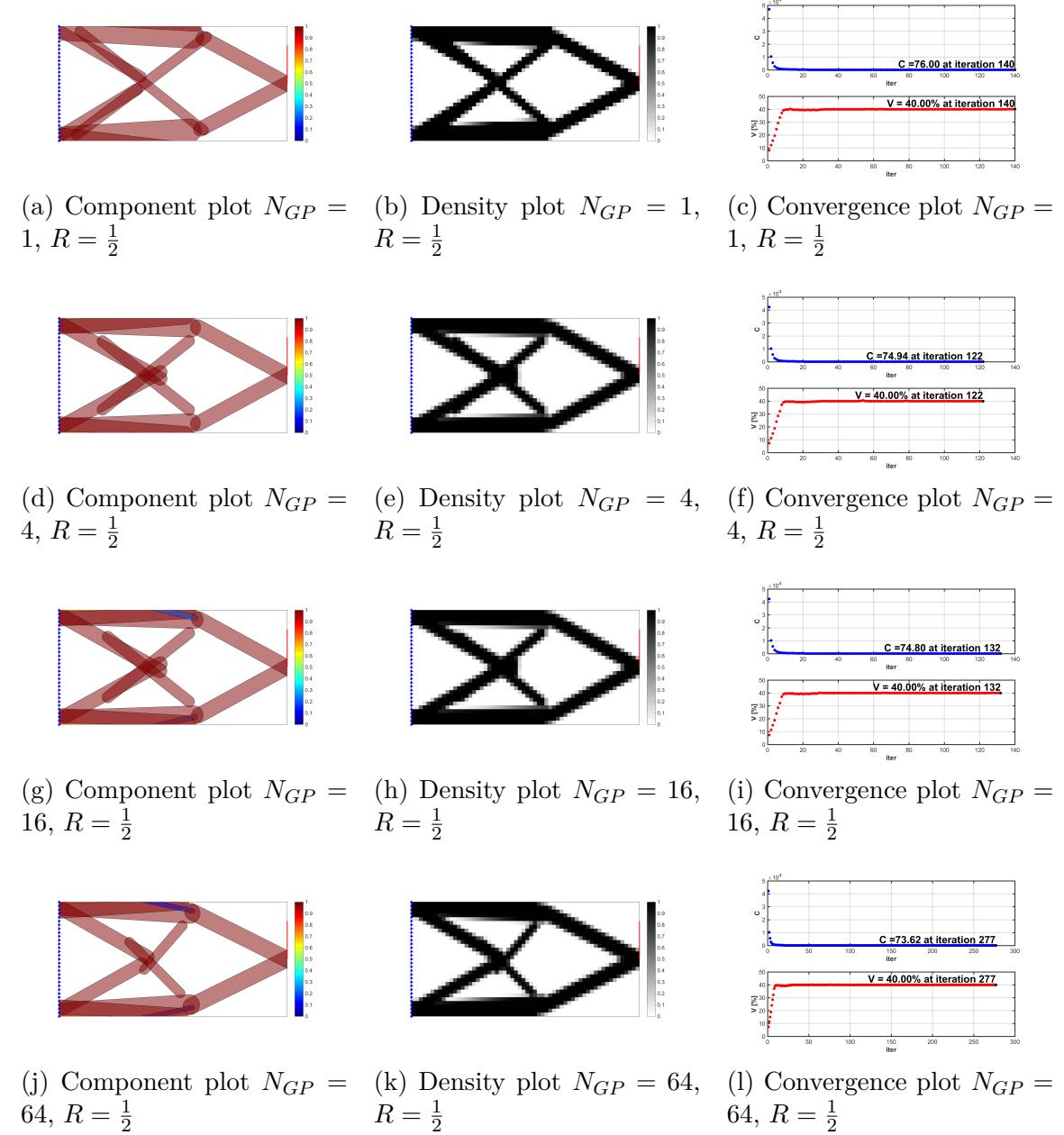


Figure 3.25: Short Cantilever Beam Topology optimization using the AGP method for variable number of Gauss points  $N_{GP}$ .

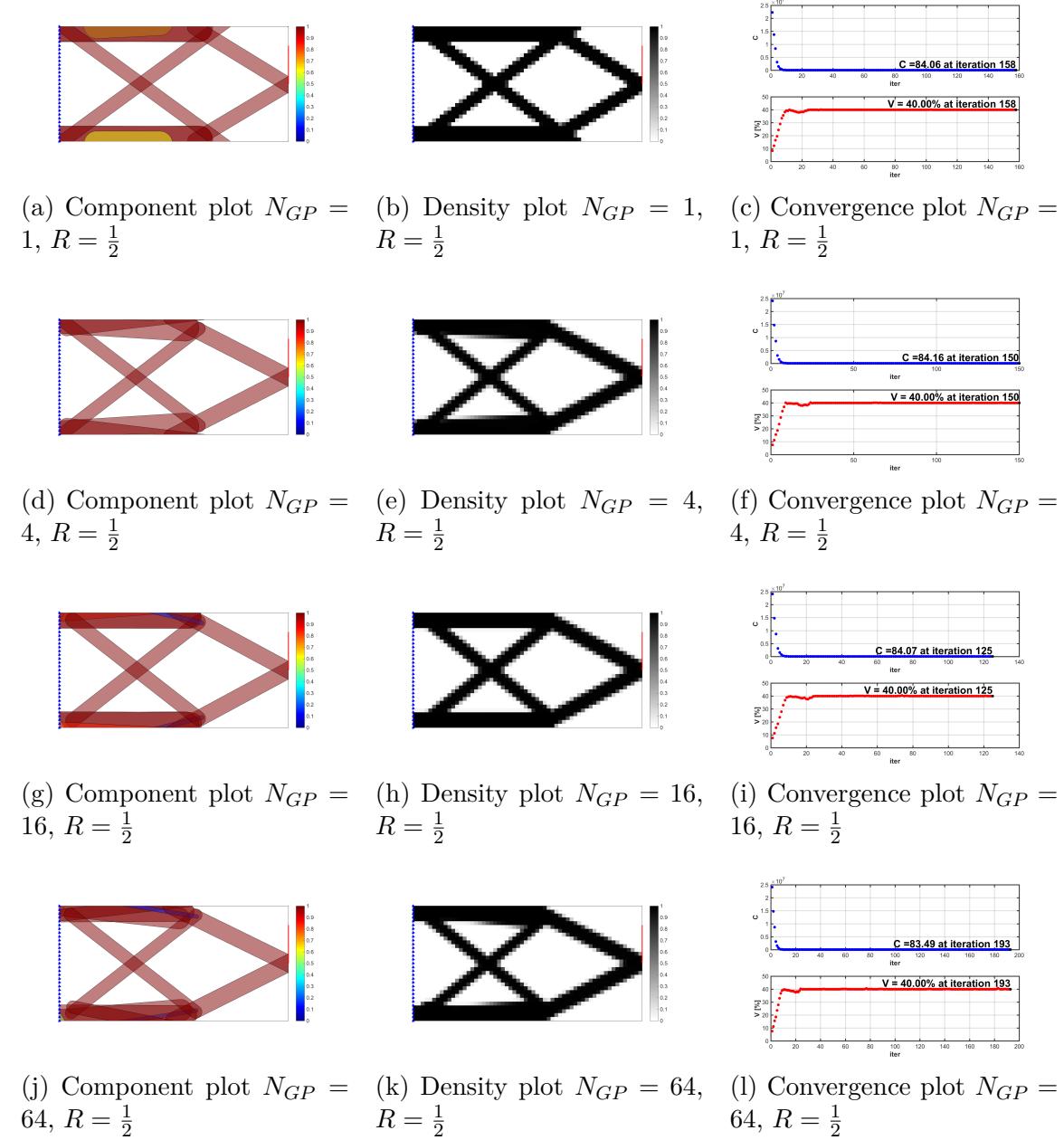


Figure 3.26: Short Cantilever Beam Topology optimization using the AMNA method for variable number of Gauss points  $N_{GP}$ .

of view this method shares the same cost for the simulation as the original problem but still requires more memory than GGP. In fact, in order to build the stiffness matrix one needs in 2D 64 terms coming from each Gauss point to be computed vs the unique value in GGP needed for the assembly. Still as for the proposed approach the shear-locking problem can have an impact on the response coming from an ill refined solution.

- **Use Generalized Geometry Projection** increasing  $N_{GP}$ . That is a very inexpensive way to attenuate mesh induced inflection points for the responses. In this way one can also consider thinner components in the optimization, with a very small additional memory and computational burden coming from the projection. It must be noted that using a clever choice of  $R$  and of  $N_{GP}$  the computational burden of adapted techniques may be controlled. In fact as is the case in the MMC approach with  $N_{GP} = 4$ ,  $R = \frac{\sqrt{3}}{2}dx$ , since the sampling window Gauss points coincide with the finite element mesh nodes, one has to compute only  $(n_{elx} + 1)(n_{ely} + 1)$  local volume fractions, instead of  $4n_{elx}n_{ely}$  required in the case of non-coincidence with other element sampling window Gauss points.
- **Change the computation of sensitivity.** In level set topology optimization approaches the issue of computing design sensitivities is widely studied. Some interesting techniques can also be adapted for the GGP approach, like the boundary integral approach [Cai et al., 2014] or the narrow-band domain integral scheme [Zhou et al., 2016].
- **Use finite differences for characteristic function gradients evaluation** and increase the perturbation step. It must be noted that in this work all Adapted approaches have been implemented using explicit evaluation of characteristic functions. It is also possible to deal with saddle points employing finite differences for the gradient of the characteristic function, as it is done in MMC [Zhang et al., 2016c]. When doing so, the optimization solver could be able to escape from saddle points. This solution comes with an increased computational burden induced by the evaluation of the characteristic function sensitivity at the Gauss points. Moreover, the choice of the finite difference represents a non-trivial tradeoff between the avoidance of optimization solver convergence to saddle points and the gradient evaluation accuracy.

### 3.9 2D stress based topology optimization using a Lagrangian approach

This section addresses stress based topology optimization using the GGP approach. Stress based topology optimization was investigated in [Zhang et al., 2017b] using the Geometry projection approach and in [Zhang et al., 2018b] using the Moving Morphable Voids approach. This section also adopted the formulation of [Verbart et al., 2017] already studied in section 2.3. The only change that one would adopt is in the design variables that this time would be

the ones of section 3.1:

$$\begin{cases} \min \{|\Omega_{el}|\}^T \{x\} \\ s.t. \\ \{l_b\} \leq \{x\} \leq \{u_b\} \\ G_{KS}^l \leq 0 \end{cases} \quad (3.112)$$

This paragraph also explores continuation strategies, and adaptive allowable update as in [Le et al., 2010, Zhang et al., 2017b]. For brevity here only MNA approach will be considered as a particular case of GGP approach. Stress based formulation being particularly nonlinear need special attention to both the optimizer and the model update with respect to the configuration. To avoid saddle point that deprecate MMA optimizer performance a large enough value of  $\epsilon$  should be considered. For similar reasons the value of penalty  $p_b$  should be kept to small enough values. After a first convergence phase  $p_b$  can be increased up to a satisfactory value (that can avoid intermediate density at component joints) and reduce the gray thickness reducing  $\epsilon$ . Moreover the stress aggregation needs a correction on the stress allowable value that here it is addressed with a continuation like strategy. As a matter of fact one can make the hypothesis that for each value of  $\sigma_{lim}$  the final von Mises stress will be at a value depending on  $\sigma_{lim}$ , the aggregation constant  $P$ , on the number of Gauss points where the stress is computed and on the particular local minimum on which the optimizer will converge. As a matter of fact, when convergence is attained the actual maximum of von Mises stress, stops changing with iterations. Thanks to that, each time convergence is reached one can make an update of the allowable that compensates for the discrepancy between the maximum von Mises stress and the desired allowable value. In order to do that one needs to build a model for the relation between the allowable and the maximum stress at convergence for a given value of  $P$ . A simple way of doing that is making the proportional hypothesis i.e:

$$\sigma_{max} = C\sigma_{lim} \quad (3.113)$$

Where  $C$  is a constant that needs to be determined:

$$C = \frac{(\sigma_{max})_{nk}}{(\sigma_{lim})_{nk}} \quad (3.114)$$

To enforce that  $(\sigma_{max})_{(n+1)k} = \sigma_{alw}$  one should then have:

$$(\sigma_{lim})_{(n+1)k} = \frac{1}{C}\sigma_{alw} = \frac{(\sigma_{lim})_{nk}}{(\sigma_{max})_{nk}}\sigma_{alw} \quad (3.115)$$

This update can be repeated until:

$$|\sigma_{alw} - \sigma_{max}| < tol_\sigma \quad (3.116)$$

where the value of the tolerance  $tol_\sigma$  is a user choice. For the initial guess design the well connected design of [Zhang et al., 2017b] (c.f. figure 3.27) was considered as precaution against convergence issues. The design domain is  $100 \times 100$  and both  $100 \times 100$  and  $200 \times 200$  meshes were considered. In table 3.4 one can find the list of parameters selected for this study, as for the study in subsection 2.3, the choice of material properties are dimensionless

and same as the one chosen in [Verbart et al., 2017] for his numerical investigation. Even in this stress analysis the load is distributed over a length of 5. For the stress aggregation

Table 3.4: Parameters used for stress based topology optimization of the L-shape

Parameter name	symbol	value
Material Young Modulus	$E$	1
Design zone width (out of plane direction)	$b$	1
Total vertical load	$F$	1
Poisson ratio	$\nu$	0.3
AMNA parameter $\gamma_v$	$\gamma_v$	1
AMNA parameter $\gamma_c$	$\gamma_c$	3
initial AMNA parameter $\varepsilon$	$\varepsilon$	6
initial AMNA parameter $p_b$	$p_b$	2
AMNA parameter $E_{min}$	$E_{min}$	$10^{-6}$
Aggregation constant for saturation	$\kappa_s$	$10^2$
Aggregation constant	$\kappa$	10
MMA moving limit		0.1
MMA initial moving limit		0.001
MMA incremental factor asyincr		1.2
MMA decremental factor asydecr		0.4
MMA parameter albefa		0.1
MMA parameter move		0.1
MMA parameter raa0		0.01
Number of Gauss Point in the sampling window	$N_{GP}$	1
Sampling window infinity norm radius	$R$	$0.5dx$
mesh size in x and y directions	$dx, dy$	$\frac{100}{n_{elx}}$
Domain sizes	$L_x, L_y$	100
Stopping criterion, design variable variation		0.001
Minimal x position	$x_{min}$	-1
Minimal y position	$y_{min}$	-1
Minimal length	$L_{min}$	0
Minimal height	$h_{min}$	6
Minimal angle	$\theta_{min}$	$-2\pi$
Minimal component density	$m_{min}$	0
Maximal x position	$x_{max}$	$n_{elx} + 1$
Maximal y position	$y_{max}$	$n_{ely} + 1$
Maximal length	$L_{max}$	$\sqrt{n_{elx}^2 + n_{ely}^2}$
Maximal height	$h_{max}$	$\sqrt{n_{elx}^2 + n_{ely}^2}$
Maximal angle	$\theta_{max}$	$2\pi$
Maximal component density	$m_{max}$	1
Allowable stress	$\sigma_{alw}$	1
Stress enforcement tolerance	$tol_\sigma$	0.01

constant  $P = 8$  was considered for the  $100 \times 100$  mesh and  $P = 16$  was used for the  $200 \times 200$

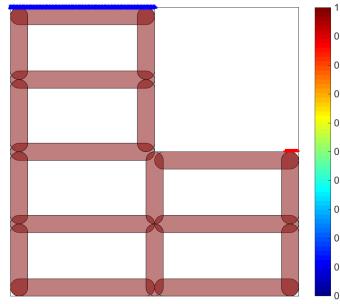


Figure 3.27: Starting guess for the L-shape stress based topology optimization. Similar to the one in [Zhang et al., 2017a]

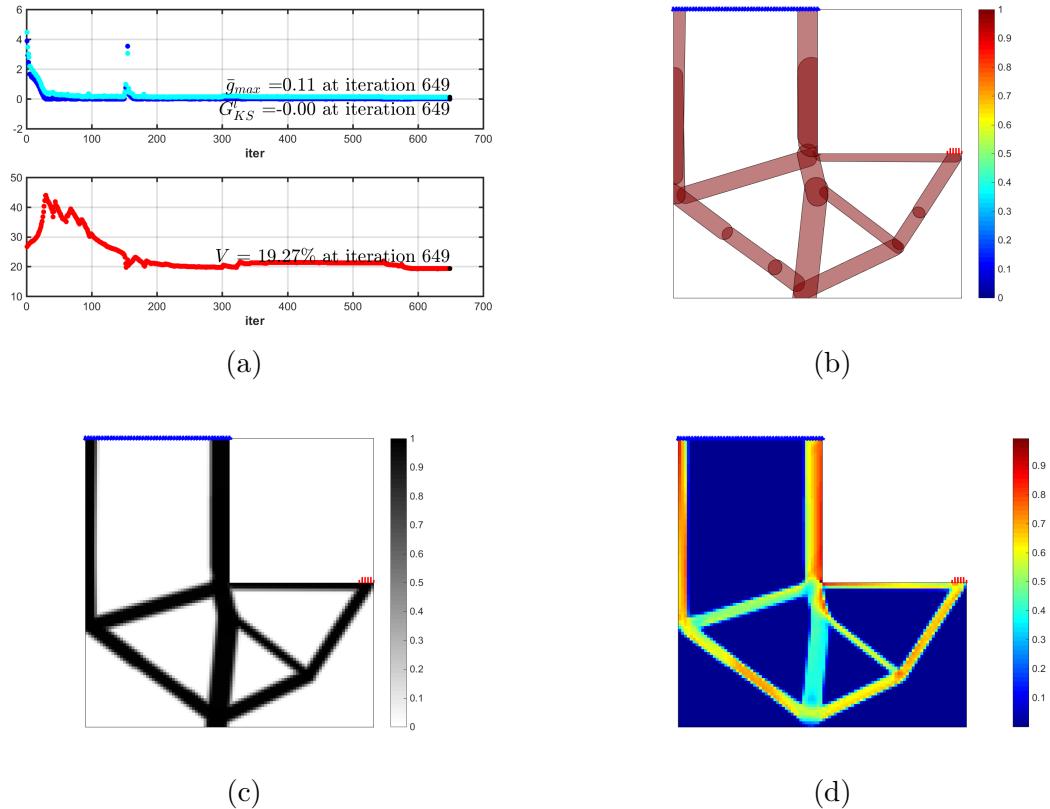


Figure 3.28: L-shape stress based topology optimization results using a  $100 \times 100$  mesh,  $P = 16$ . (a) convergence history, (b) component plot at the solution, (c) corresponding density distribution , (d) macroscopic stress distribution

mesh. For the continuation strategy, the value of  $p_b$  was initially set to 2 and then increased to 3 after 300 iterations or when convergence was reached.  $\varepsilon$  was initially set to 6 and then decreased by 1 each 300 iterations or when convergence was reached. The continuation of  $\varepsilon$  was stopped when  $\varepsilon = 3$ . The value of  $h_{min}$  was also updated following the same history of

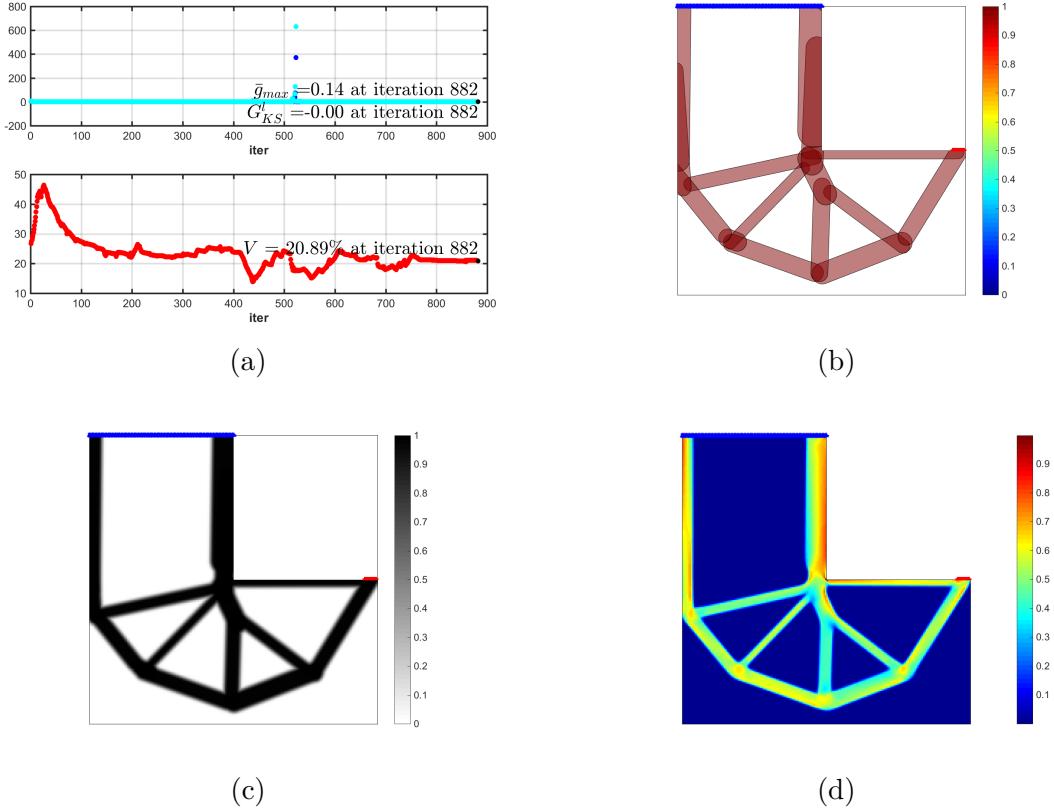


Figure 3.29: L-shape stress based topology optimization results using a  $200 \times 200$  mesh,  $P = 16$ . (a) convergence history, (b) component plot at the solution, (c) corresponding density distribution , (d) macroscopic stress distribution

$\varepsilon$ . Finally for the  $\sigma_{lim}$  update, this was started at the value of 1 and then changed every 20 iterations according to equation (3.115). The stress limit update frequency was selected as a trade between the iterations required for MMA to be sufficiently converged while reducing the overall convergence history. The update was stopped when equation (3.116) was satisfied. Topology optimization results are reported in figure 3.28 and 3.29 for both the  $100 \times 100$  and the  $200 \times 200$  meshes. The maximum macroscopic stress is enforced to be equal to the allowable value of 1 in both results (c.f. figures 3.28d, 3.29d). Continuation and limit stress update influence in both cases the convergence history (c.f. figures 3.28a, 3.29a). Comparing these results with the ones of SIMP approach obtained changing the value of  $\sigma_{lim}$  c.f. figures 2.23c, 2.23d, 2.23f one can observe that the final mass is slightly improved. On top of that the iterations needed to converge to nearly black and white designs (c.f. figures 3.28c, 3.29c) are reduced compared to the one needed by SIMP with continuation. Finally the explicit geometric description of solutions is available (c.f. figures 3.28b, 3.29b) and this is an advantage for the next step in the design validation process. Being the geometric description available one can in fact re-mesh the solution and make further checks on the solution. On the other hand both solutions do not eliminate the stress concentration at the inner corner. These solutions could further be improved to achieve singular optima that eliminate the stress

singularity at the inner corner. These could be achieved changing optimization parameters such as starting guess or MMA moving limit.

## 3.10 MNA application to the pylon and engine mount design

This section develops the MNA approach for 3D applications, to be intended for the engine mounts and pylon design already addressed in section 2.4. Explicit geometric components are built from simple geometric primitives that are combined together. In subsection 3.10.1 the relation between convex primitives and local volume fraction is established. Boolean operations, i.e. union intersection and logical subtraction between components are then defined to build more complex 3D geometric primitives. In subsection 3.10.2 a way of enforcing the components to stay inside the convex hull of the design region is explained. Finally subsection 3.10.3 and 3.10.4 show the industrial application of MNA to the topology optimization problem.

### 3.10.1 3D geometric components

Here a library of simple convex 3D components is presented. For each of this component a mathematical geometry description is established. The link to the smooth Heaviside function is then introduced as a Boolean operation.

#### 3.10.1.1 Rectangular parallelepiped components

A rectangular parallelepiped component is shown in figure 3.30, with its local reference frame and design variables. The configuration of such design can be fully described by the 9 design variables:  $\{X, Y, Z, \theta_x, \theta_y, \theta_z, L_\xi, L_\eta, L_\zeta\}$ . Where  $X, Y$  and  $Z$  are the coordinates of origin of the reference frame  $O_c\xi\eta\zeta$  defined in the center of the component and with axes oriented along 3 orthogonal edges of the component.  $\theta_x, \theta_y$  and  $\theta_z$  are rotation angles needed to change from the reference frame  $Oxyz$  to the reference frame  $O_c\xi\eta\zeta$ . Finally the sizes of the component along each local coordinate  $\xi, \eta, \zeta$  are respectively  $L_\xi, L_\eta$  and  $L_\zeta$ . A rigorous mathematical description of such component can be stated as:

$$\mathcal{P} \equiv \left\{ \{X_g\} = \{x, y, z\}^T : \left| -\frac{L_\xi}{2} \leq \xi \leq \frac{L_\xi}{2} \cap -\frac{L_\eta}{2} \leq \eta \leq \frac{L_\eta}{2} \cap -\frac{L_\zeta}{2} \leq \zeta \leq \frac{L_\zeta}{2} \right. \right\} \quad (3.117)$$

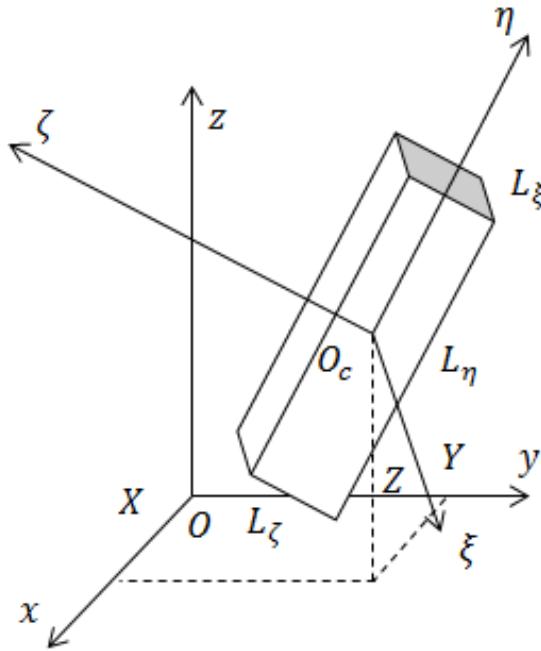


Figure 3.30: Parallelepiped component, its local reference frame and design variables.

In other words the rectangular parallelepiped can be represented as the intersection of six half spaces whose equations can be also written in the form:

$$v_1 = -\frac{L_\xi}{2} - \xi \leq 0 \quad (3.118)$$

$$v_2 = -\frac{L_\xi}{2} + \xi \leq 0 \quad (3.119)$$

$$v_3 = -\frac{L_\eta}{2} - \eta \leq 0 \quad (3.120)$$

$$v_4 = -\frac{L_\eta}{2} + \eta \leq 0 \quad (3.121)$$

$$v_5 = -\frac{L_\zeta}{2} - \zeta \leq 0 \quad (3.122)$$

$$v_6 = -\frac{L_\zeta}{2} + \zeta \leq 0 \quad (3.123)$$

where \$v\_i\$ can be identified as the signed distance from each parallelepiped face. In this way equation (3.117) can be written more compactly as:

$$\mathcal{P} \equiv \left\{ \{X_g\} = \{x, y, z\}^T : \left| \bigcap_{i=1}^6 v_i \leq 0 \right. \right\} \quad (3.124)$$

### 3.10.1.2 Cylinder components

A cylinder component is shown in figure 3.31, with its local reference frame and design variables. The configuration of that component can be fully described by the 7 design variables:  $\{X, Y, Z, \theta_x, \theta_y, \theta_z, R, L_\zeta\}$ . Where  $X, Y$  and  $Z$  are the coordinates of origin of the reference frame  $O_c\xi\eta\zeta$  defined in the center of the component and with axes  $\zeta$  oriented along the cylinder axis.  $\theta_x, \theta_y$  and  $\theta_z$  are rotation angles needed to change from the reference frame  $Oxyz$  to the reference frame  $O_c\xi\eta\zeta$ . Finally the sizes of the component are given by cylinder height  $L_\zeta$  and by its radius  $R$ . Again a more rigorous definition is:

$$\mathcal{C} \equiv \left\{ \{X_g\} = \{x, y, z\}^T : | \sqrt{\xi^2 + \eta^2} \leq R \cap -\frac{L_\zeta}{2} \leq \zeta \leq \frac{L_\zeta}{2} \right\} \quad (3.125)$$

A cylinder is therefore seen as the intersection of 3 inequalities, which can also be written

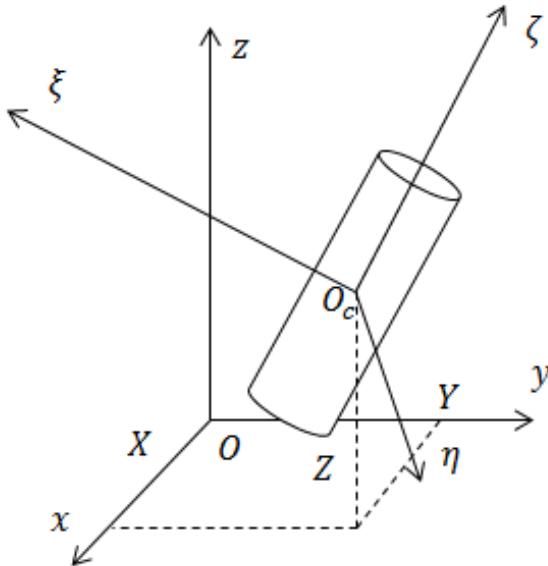


Figure 3.31: Parallelepiped component, its local reference frame and design variables.

as:

$$v_1 = -R + \sqrt{\xi^2 + \eta^2} \leq 0 \quad (3.126)$$

$$v_2 = -\frac{L_\zeta}{2} - \zeta \leq 0 \quad (3.127)$$

$$v_3 = -\frac{L_\zeta}{2} + \zeta \leq 0 \quad (3.128)$$

So that equation (3.125) can be compactly written as:

$$\mathcal{C} \equiv \left\{ \{X_g\} = \{x, y, z\}^T : \left| \bigcap_{i=1}^3 v_i \leq 0 \right. \right\} \quad (3.129)$$

### 3.10.1.3 Polyhedral components

It is natural to extend the mathematical description of a rectangular parallelepiped component to a general polyhedral component. On the other hand definition such as equations (3.125) and (3.129) cannot be extended to general polyhedral component. Concave polyhedrons cannot in fact be written as the intersection of inequalities but may require also union subspaces. For convex polyhedral ( c.f. figure 3.32) on the other hand one can always build signed distance from each facet as:

$$v_i = \{XgP_i\}^T \{n_i\} \quad \forall i = 1, 2, \dots, N_F \quad (3.130)$$

Where  $P_i$  is a point belonging to the  $i^{th}$  polyhedron facet and  $\{n_i\}$  is the corresponding outward unit vector and  $N_F$  is the number of facets of the polyhedron. Given these definitions, the mathematical description of a Polyhedron can be written as:

$$\mathcal{Q} \equiv \left\{ \{X_g\} = \{x, y, z\}^T : \left| \bigcap_{i=1}^{N_F} v_i \leq 0 \right. \right\} \quad (3.131)$$

A polyhedron can be described by the plane of the facets. This means that the number of

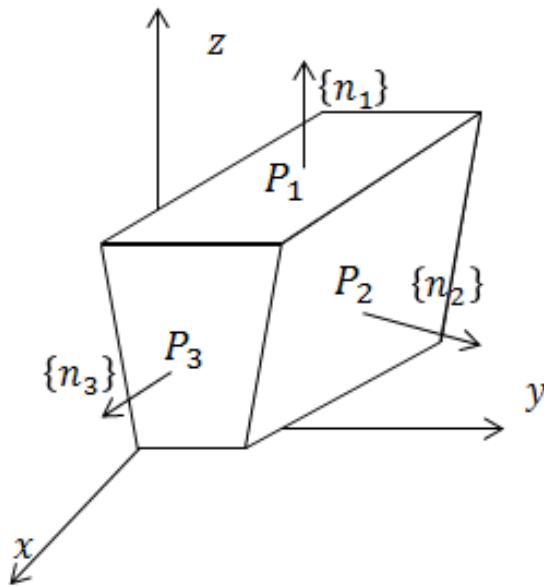


Figure 3.32: Convex polyhedral component scheme

design variables associated with such a component are  $3 \times N_F$ : 2 angles for the orientation of  $\{n_i\}$  and the distance of  $P_i$  from the origin in the direction of  $\{n_i\}$ . To determine the mathematical description of concave polyhedrons one needs first to consider a concave polygon to be described using signed distances (c.f. figure 3.33). Concave shapes have at least a couple of sides that belong to the positive half spaces defined by the signed distance:

$$\{P_l P_k\}^T \{n_k\} > 0 \quad (3.132)$$

$$\{P_k P_l\}^T \{n_l\} > 0 \quad (3.133)$$

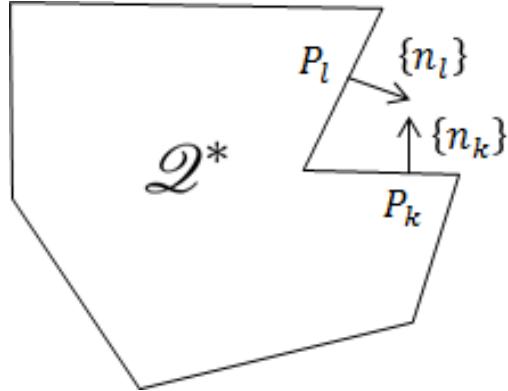


Figure 3.33: Concave polygon scheme for the signed distance based description

When this is the case in order to avoid the elimination of a part of the shape one needs to consider the logic union of these two conditions. In fact the polygon of figure 3.33 can be described as:

$$\mathcal{Q}^* \equiv \left\{ \{X_g\} = \{x, y, z\}^T : |(v_l \leq 0 \bigcup v_k \leq 0) \bigcap (\bigcap_{i \neq l, k} v_i \leq 0) \right\} \quad (3.134)$$

In other words, for general polygons and polyhedrons, when there are groups of facets that satisfy for each couple of facets in the group both equations (3.132) and (3.133), then in equation (3.131) each corresponding inequality should be replaced by the union of the conditions of facets belonging to the same group. More formally given the groups :

$$\mathcal{G}_j \equiv \left\{ s \in \mathbb{N} : | 1 \leq s \leq N_F, \forall k \neq l \in \mathcal{G}_j \quad (\{P_l P_k\}^T \{n_k\})(\{P_k P_l\}^T \{n_l\}) > 0 \right\} \quad (3.135)$$

Given the complementary ensemble as:

$$\bar{\mathcal{G}} \equiv \{1, 2, 3, \dots, N_F\} \setminus \bigcup_{j=1}^{N_c} \mathcal{G}_j \quad (3.136)$$

Where \$N\_c\$ is the number group of the kind of \$\mathcal{G}\_j\$ that can be determined for the polygon. Given the half spaces defined as:

$$\mathcal{S}_j \equiv \left\{ \{X_g\} = \{x, y, z\}^T : | \bigcup_{i \in \mathcal{G}_j} v_i \leq 0 \right\} \quad (3.137)$$

The mathematical definition of the concave polyhedron is given by:

$$\mathcal{Q}^*_j \equiv \left\{ \{X_g\} = \{x, y, z\}^T : |(\bigcap_{i=1}^{N_c} \mathcal{S}_i) \bigcap (\bigcap_{i \in \bar{\mathcal{G}}} v_i \leq 0) \right\} \quad (3.138)$$

### 3.10.1.4 Activation function and geometric intersection

Convex components like rectangular parallelepipeds, cylinders and convex polyhedrons can be described as the intersection of half spaces. By this definition it is possible to build in a simple manner the smooth Heaviside function needed for the use of Generalized Geometry Projection. First it is possible to define the activation function  $W(v)$ :

$$W(v, g_t) = \begin{cases} 1 & \text{if } v \leq -g_t \\ -\frac{3}{16} \left(\frac{v}{g_t}\right)^5 + \frac{5}{8} \left(\frac{v}{g_t}\right)^3 - \frac{15}{16} \left(\frac{v}{g_t}\right) + \frac{1}{2} & \text{if } -g_t < v < g_t \\ 0 & \text{otherwise.} \end{cases} \quad (3.139)$$

This function is equal to 1 only for negative input and smoothly passes to 0 values for positive input in the transition region between for  $-g_t < v < g_t$ . The only difference with respect to function in (3.30)  $\in C^1(-\infty, \infty)$  is that the function in (3.139)  $\in C^2(-\infty, \infty)$  with respect to the input  $v$  and that the transition region is centered in 0. One can observe that the activation function of the intersection of several conditions can be obtained, again treating the  $W$  as logical value. One could at this point use approaches analogous to the one introduced in subsection 3.6. For brevity here only Boolean assembly will be considered i.e:

$$W_{\mathcal{P}} = \prod_{i=1}^6 W(v_i, g_t) \quad (3.140)$$

$$W_{\mathcal{C}} = \prod_{i=1}^3 W(v_i, g_t) \quad (3.141)$$

$$W_{\mathcal{Q}} = \prod_{i=1}^{N_F} W(v_i, g_t) \quad (3.142)$$

When considering the union of subspaces one can adopt one of the approaches discussed in subsection 3.6. One can observe that using such functions it is still possible to evaluate analytically all derivatives, simply adopting chain rules. For concave polygons both intersection and union should be applied to activation functions i.e. :

$$W_{\mathcal{Q}^*} = \left( \prod_{i \in \mathcal{G}} W(v_i, g_t) \right) \left( \prod_{j=1}^{N_c} \left( 1 - \prod_{i \in \mathcal{S}_j} (1 - W(v_i, g_t)) \right) \right) \quad (3.143)$$

### 3.10.2 Convex hull constraint

In the previous explicit optimization results 3.8.2 one can observe that in the optimized configuration components can lie outside the design region. This behavior is induced by the fact that one density and Young's modulus layout can be corresponding to several configurations. This is not a problem from a mathematical point of view, but in some cases for a complex design zone, can produce a complex geometry that is the intersection between components and the design region. To avoid such issue in [Zhang et al., 2018a] the authors proposed an

approach that put ghost nodes outside the design region and imposed the projected volume fraction to be smaller than a given tolerance. With such an approach all components are enforced to stay inside the design domain. Such approach hides some difficulties like, the generation of the ghost nodes and the choice of their relative distance. Here an alternative that tries to use directly the information coming from the geometry of the design region was proposed. This approach applies to convex polygonal design regions even if it is possible to extend it to concave design zone. Given the design region finite element mesh it is possible to consider its convex hull as the smallest convex polygon in which the design region lies. Its determination can be obtained using the convhull Matlab native function for example. Once this is done the convex hull surface triangulation can be exploited to generate signed distances with respect to the outward normal vector as described in figure 3.34b. Given the control point  $P_i$  it is possible to say that it is inside the convex hull as long as all the signed distances from the convex hull facets are negative. This can be computed for the  $k^{th}$  triangle of the triangulation of the convex hull and for the  $i^{th}$  control point as:

$$d_{ik} = \{A_k P_i\}^T \{n_k\} \quad (3.144)$$

where  $A_k$  is a vertex of the triangle (c.f. figure 3.34b).  $P_i$  have to be chosen in order to enforce that the components lies inside the design region. For rectangular parallelepiped components the eight vertices can be used for this purpose. In order to have that all components lie inside the design region one must have that

$$\max_{i=1,2,\dots,N_p} \max_{k=1,2,\dots,N_T} d_{ik} \leq 0 \quad (3.145)$$

Where  $N_p$  is the total number of control points and  $N_T$  is the total number of triangles in the convex hull triangulation. Once again maximum function should be avoided for differen-

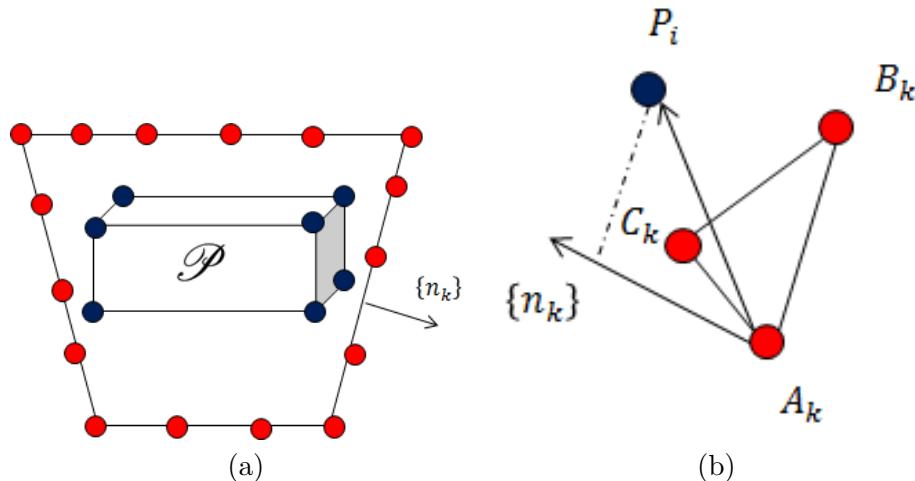


Figure 3.34: Convex hull constraint scheme, a) in blue Parallelepiped control points, in a red design region convex hull nodes; b) scheme for the signed distance between the control point and a triangular facets of the convex hull

tiability issues. A Kreisselmeier-Steinhauser approximation [Kreisselmeier and Steinhauser, 1980] of the maximum can be adopted to enforce this constraint :

$$G_{ch} = \frac{1}{P_{ch}} \log \left( \frac{1}{N_T N_P} \sum_{i=1}^{N_P} \sum_{k=1}^{N_T} e^{P_{ch} d_{ik}} \right) \leq 0 \quad (3.146)$$

where  $P_{ch}$  is the aggregation constant for such constraint. As for the stress aggregation when using the lower bound aggregation, the actual constraints could be violated. This is, by the way, acceptable especially near regions where the boundary conditions are applied. In this way the actual density at boundary conditions can be 1.

### 3.10.3 Rectangular parallelepiped component topology optimization

This section illustrates the application of the MNA approach with 3D rectangular parallelepiped components, symmetric and with 3 load cases. To enforce symmetrical solutions each component was always projected together with its symmetrical with respect to the xz plane. The formulation of the optimization problem is almost identical to the one of equation 2.100, the design variables are this time the explicit design variables of the rectangular parallelepiped components. Moreover the convex hull constraint was considered in the optimization (3.146):

$$\begin{cases} \min_{\{X\}} V(\{X\}) \\ s.t. \\ \{l_b\} \leq \{X\} \leq \{u_b\} \\ G_T = \frac{\frac{1}{P_l} \log \left( \frac{\sum_{l_c=1}^{N_l} e^{P_l (\Delta TSFC\%) l_c}}{N_l} \right) - T_0}{T_0} \times 100 \leq 0 \\ [K(\{X\})][U] = [F(\{X\})] \\ \frac{1}{P_l} \log \left( \frac{\sum_{l_c=1}^{N_l} e^{P_l (G_{KS}^l) l_c}}{N_l} \right) \leq 0 \\ G_{ch} \leq 0 \end{cases} \quad (3.147)$$

where  $\{X\}$  indicates the design variable vector. The starting guess was selected to be well connected to avoid convergence issues. The component densities are not considered as variables for this optimization. The problem set up parameters are detailed in table 3.5<sup>8</sup>. The initial design is chosen to be well connected. For this purpose some points on the region where the engine is connected to the design region and where boundary conditions are applied, were connected between each other by components with constant  $L_\xi$  and  $L_\eta$  as shown in figure 3.35. Convergence history is shown in figure 3.36 for Volume fraction for the aggregation function for stress, for the aggregated performance and for  $G_{ch}$  (c.f. equation (3.146)). The final design respects all constraints and is shown in figure 3.37. As expected part of the components lay outside the convex hull represented in gray in the same figure. Still it is pos-

<sup>8</sup>  $x_{min}, y_{min}, z_{min}$  and  $x_{max}, y_{max}, z_{max}$  represent lower and upper bounds of nodal coordinates respectively

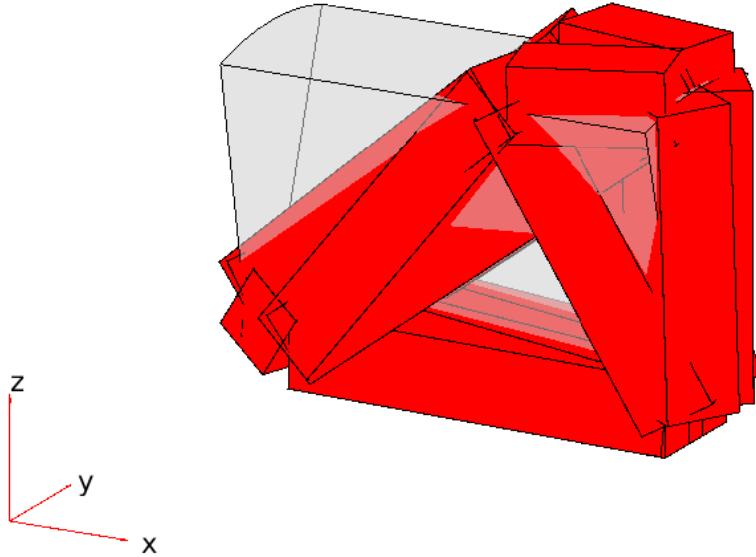


Figure 3.35: Initial guess component plot

sible to understand the solution features as in a truss pylon. In figure 3.38 the density field corresponding to the solution is represented for a threshold of 0.45. Finally displacements and stress are represented for the 3 load cases in figure 3.39. In the same figure one can also understand which load case is responsible for the stress limit activation on each component. The frontal connection between engine core and front wing mounts bear lateral load mainly, the rear part on the other hand bears both the vertical and axial loads on the engine. The V structure that connects the frontal to the rear engine casing can reduce the stresses at the interface with the engine casing and redistribute the load on a larger area. This engine stiffening structure is also added probably to increase the engine bending stiffness and reduce casing out of roundness deformations. The solution can be interpreted as an assembly of beam components with rectangular profile with the exception of the component that stiffens the engine (this time more complex as they intercept the design region boundaries). The manufacturing process for such solution and the modifications needed to attain a solution ready for production are smaller if compared to the solution of figure 2.33. In fact straight bars can be manufactured and then assembled together with fastener connection for example. The next subsection will show that this process of solution simplification can also be extended to primitives that are more similar to existent products and that are therefore prone to include other explicit constraint on the final product geometry, still keeping the great versatility of topology optimization for making components disappear.

Table 3.5: Optimization problem set up. The hypothesis made to get numerical results for the 3D MNA topology optimization on the multiple load case are listed here

Symbol	Name	Value
$E_0$	Young's Modulus	210 GPa
$\nu$	Poisson Ratio	0.29
$N_{el}$	Number of elements in the design zone	486400
$N_{DOFs}$	Number of rows of the stiffness matrix	1529847
$N_G$	Number of Gauss points in the design zone	3891200
$N_{var}$	Number of Design variables	90
$p_b$	MNA penalty	3
$g_t$	MNA gray thickness	$4.8 \times$ average mesh size
$\kappa$	IE geometric assembly aggregation constant	10
$P$	Aggregation constant for the stress constraint	4
$P_l$	Aggregation constant for load case	10
$\sigma_{lim}$	Allowable stress to be used in (2.58)	10 MPa
$\sigma_{alw}$	Maximum local allowable stress	49 MPa
$T_0$	Allowable consumption variation	0.17 %
$\{\rho f\}$	Inertial load	{0}
$X_{min}$	minimum of X position for the parallelepiped center	$x_{min}$
$Y_{min}$	minimum of Y position for the parallelepiped center	$y_{min}$
$Z_{min}$	minimum of Z position for the parallelepiped center	$z_{min}$
$(L_\xi)_{min}$	minimum of length in $\xi$ direction	0
$(L_\eta)_{min}$	minimum of length in $\eta$ direction	0
$(L_\zeta)_{min}$	minimum of length in $\zeta$ direction	0
$(\theta)_{min}$	minimum of Rotation angles in each direction	$-\pi$
$X_{max}$	maximum of X position for the parallelepiped center	$x_{max}$
$Y_{max}$	maximum of Y position for the parallelepiped center	$y_{max}$
$Z_{max}$	maximum of Z position for the parallelepiped center	$z_{max}$
$(L_\xi)_{max}$	maximum of length in $\xi$ direction	$2(x_{max} - x_{min})$
$(L_\eta)_{max}$	maximum of length in $\eta$ direction	$2(y_{max} - y_{min})$
$(L_\zeta)_{max}$	maximum of length in $\zeta$ direction	$2(z_{max} - z_{min})$
$(\theta)_{max}$	maximum of Rotation angles in each direction	$\pi$
Stopping condition on the Karush-Kuhn Tucker residual norm		$KKTn \leq 10^{-3}$
Stopping condition on the iteration number		$iter \geq 300$
MMA external move limit for first 2 iterations		0.4
MMA maximum asymptote distance from the current point		0.1

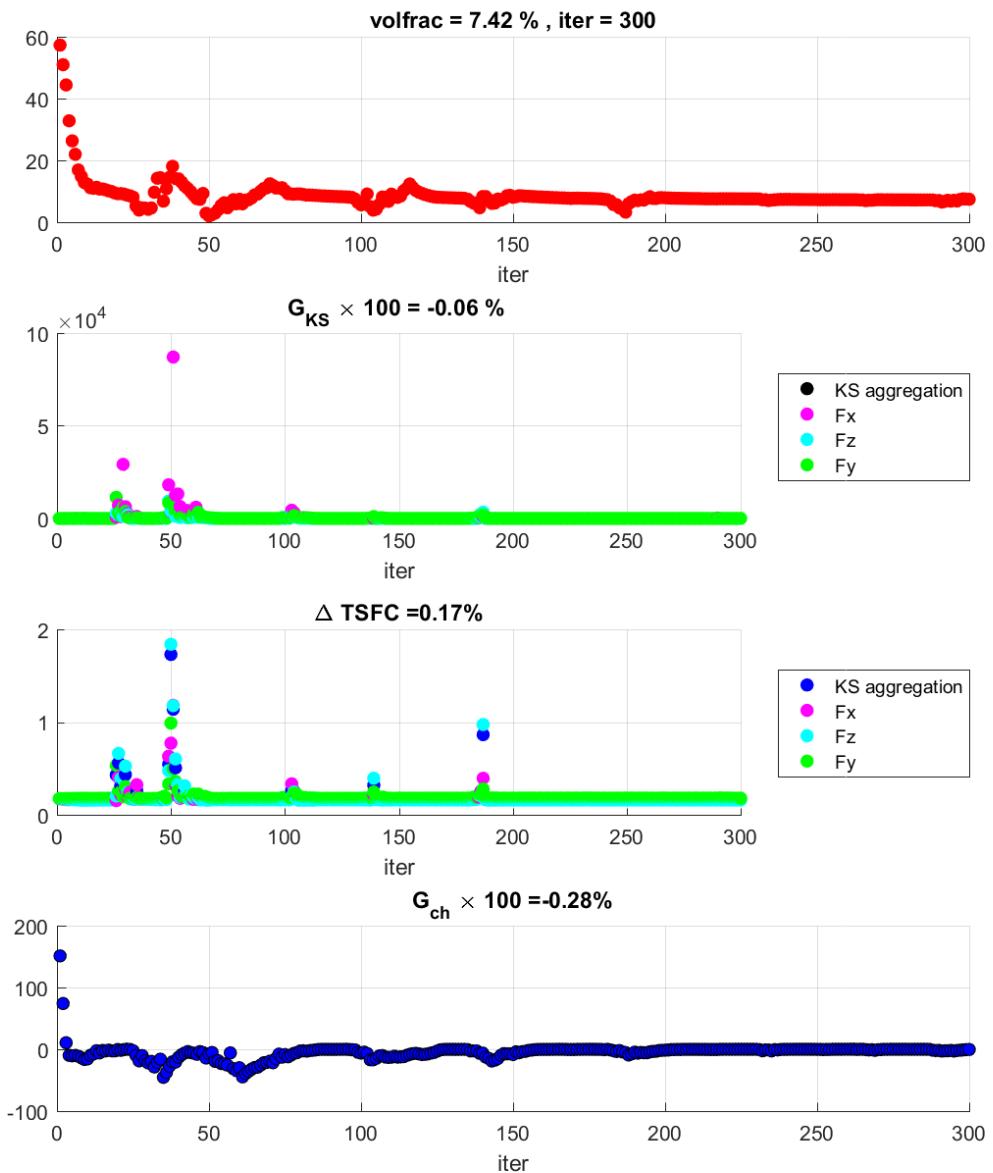


Figure 3.36: Convergence history of the multi load case topology optimization using MNA with rectangular parallelepiped

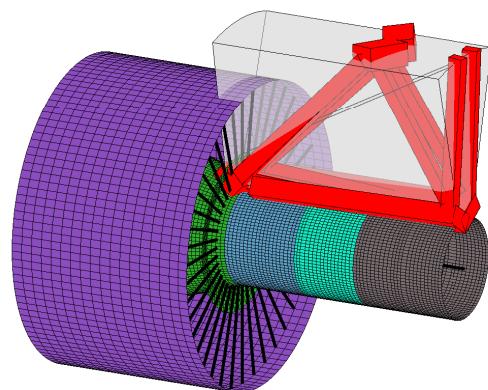


Figure 3.37: Component plot of the solution of MNA with rectangular parallelepiped components.

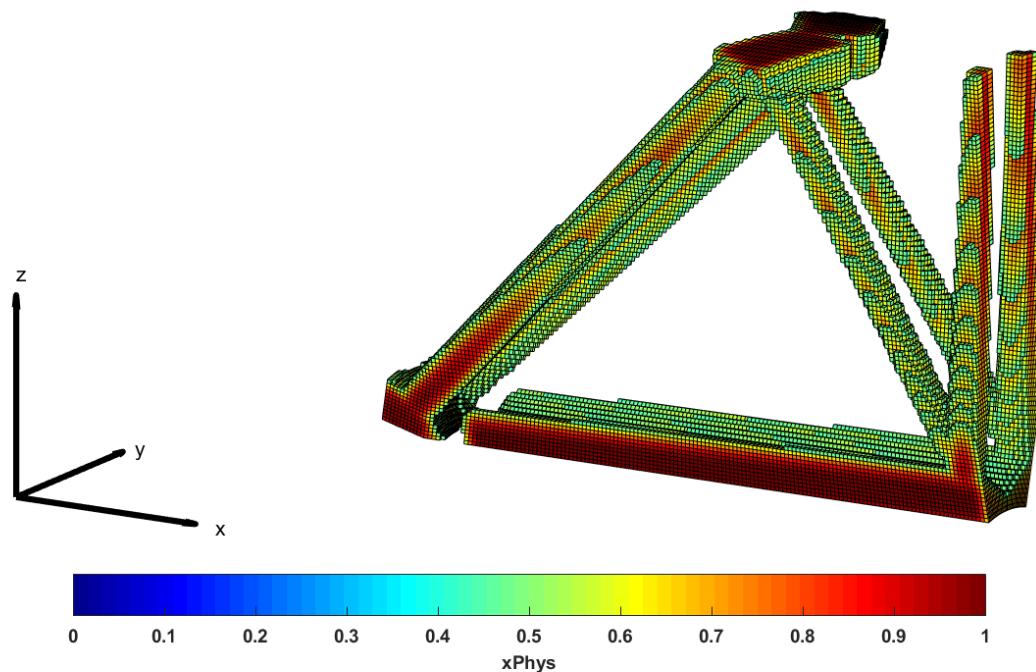


Figure 3.38: Density plot of the solution of the multi load case analysis for the pylon and engine mount design using MNA 3D with rectangular parallelepiped components. Only elements with a physical density greater than 0.45 are represented.

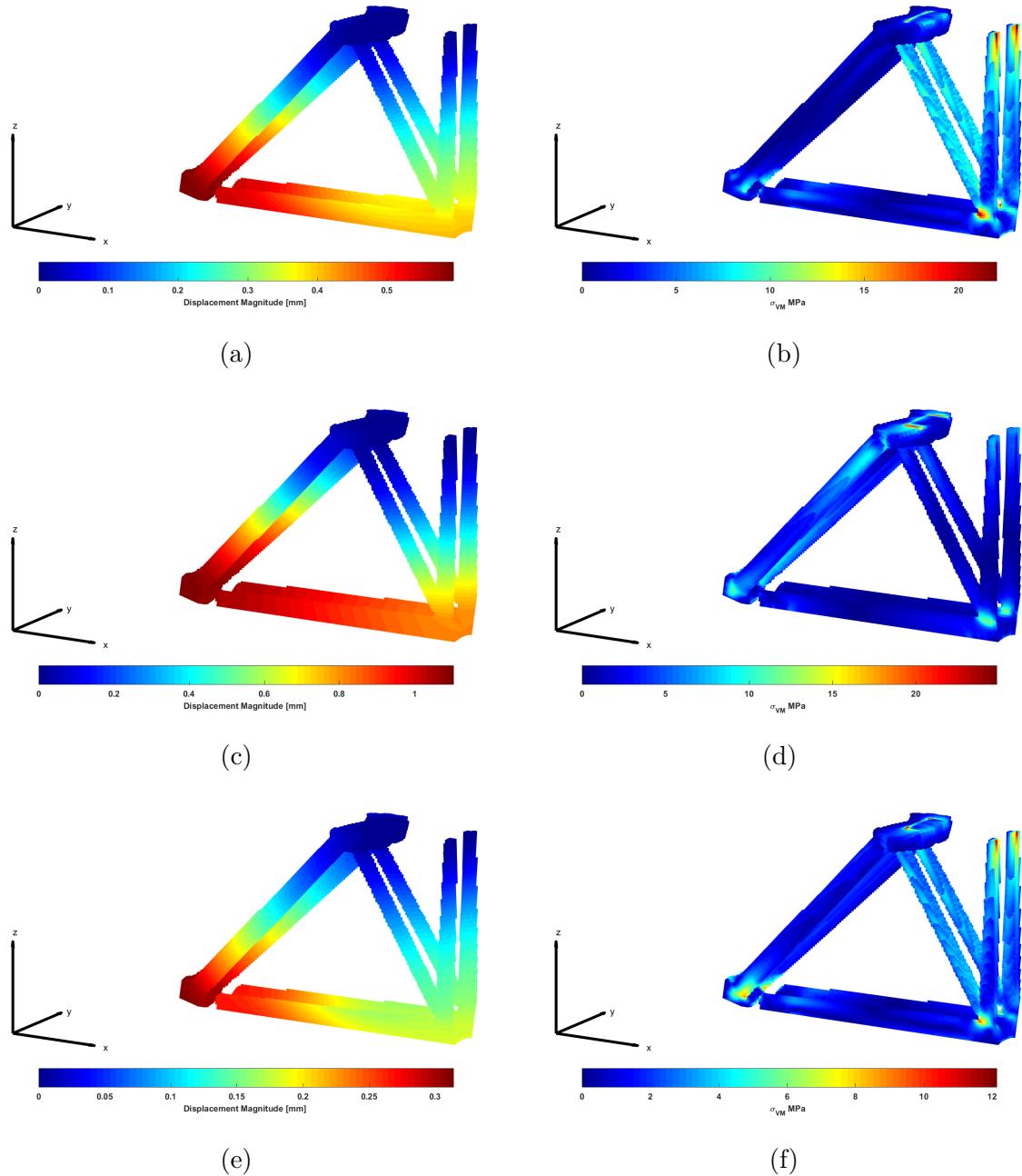


Figure 3.39: Topology optimization results for MNA with rectangular parallelepiped components. Displacement magnitude color map (a) for  $F_x$ ,(c) for  $F_y$ ,(e) for  $F_z$  respectively. von Mises stress color map (b) for  $F_x$ ,(d) for  $F_y$ ,(f) for  $F_z$  respectively.

### 3.10.4 Pylon box, engine and wing mounts design optimization

Engine pylon main structure is often made of assemblies of panels that are supposed to withstand different load conditions. Such geometry can be represented as multiple boxes concatenated the one with the other. If a simple box is considered, this can be represented as the difference of two convex hexahedra. With reference to figure 3.40, the convex hexahedron with vertexes from 1 to 8 is denoted as  $\mathcal{Q}_e$  and the one with vertexes from 9 to 16 is denoted as  $\mathcal{Q}_i$ , it is also possible to observe that the box component  $\mathcal{B}$  simply is:

$$\mathcal{B} = \mathcal{Q}_e \setminus \mathcal{Q}_i \equiv \mathcal{Q}_e \cap \bar{\mathcal{Q}}_i \quad (3.148)$$

Where  $\bar{\mathcal{Q}}_i$  indicates the complementary of ensemble of  $\mathcal{Q}_i$  of the  $\mathbb{R}^3$  space. The smooth characteristic function corresponding to the complement can be computed as:

$$W_{\bar{\mathcal{Q}}_i} = 1 - W_{\mathcal{Q}_i} \quad (3.149)$$

so that the characteristic function of the box component can be evaluated as:

$$W_{\mathcal{B}} = W_{\mathcal{Q}_e}(1 - W_{\mathcal{Q}_i}) \quad (3.150)$$

This component could be represented with the strategy explained in subsection 3.10.1.3 that

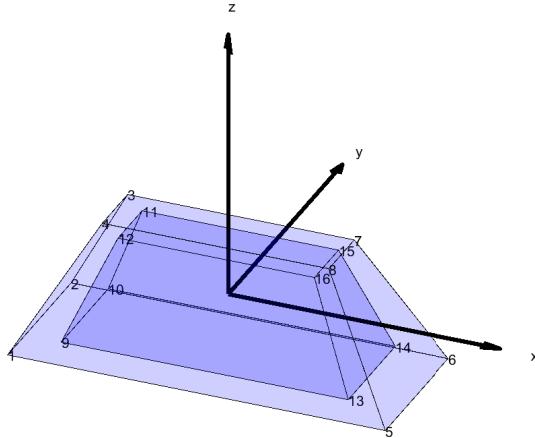


Figure 3.40: Box component example. The external and the internal hexahedra have only planar facets. The internal hexahedron vertexes are obtained translating the external hexahedron vertexes using box thicknesses.

employs the angles of the outward vector and a distance from a constant point in the same direction of each facet of the polyhedron. This strategy is non compatible with classic pylon sizing and description. Explicit sizes will be employed for the pylon box definition. These can be bounded in order to have reasonable designs. For this purpose it is worth to introduce the following notations. Lateral panels refer to the facets that are opposite along y direction. Lower and upper spar refer to the panels that are opposite in z direction and whose normal point downward and upward respectively and ribs refers to the panels that are opposite in x direction. In a pylon box one can adopt the rib center position, rib orientation angle and

sizes as design variables of the optimization (c.f. figure 3.41). As only symmetric design with respect to the  $xz$  plane should be considered some parameters like the ribs center  $Y_s$  and rotation angles around  $x$  and  $z$  ( $\theta_x$  and  $\theta_z$ ) have to be 0. The remaining parameters that can vary for each section are represented in figure 3.41 and can be stored in the design variable vector  $\{x_s\} = \{X_s, Z_s, \theta_y, h_s, b_l, b_u\}$ . There will be one of these vectors for each rib of the pylon boxes. As each pylon box shares ribs with its neighborhoods, one can say that for  $n_b$  boxes, one will have  $n_b + 1$  ribs to be parametrized, that means  $6 \times (n_b + 1)$  design variables. Actually this parametrization allows for the description of non-planar lateral panels. This is not desirable for manufacturing complexity. To avoid such situation the design variable  $b_u$  are free to vary only on one rib (for example the most forward one). For the remaining it can be determined explicitly in order to keep the lateral panel planar. For this reason the count of design variables becomes of  $5n_b + 6$ . The thickness of each panel should then be varied independently. Again for the symmetry with respect to the  $xz$  plane lateral panels are constrained to have the same thickness, which means that one can vary 5 thicknesses per box (i.e.  $\{x_b\} = \{t_1, t_2, t_3, t_4, t_5\}$ ). The total number of design variables per box is then  $10n_b + 6$ . For the engine and wing mounts connection one can also consider components that are closer

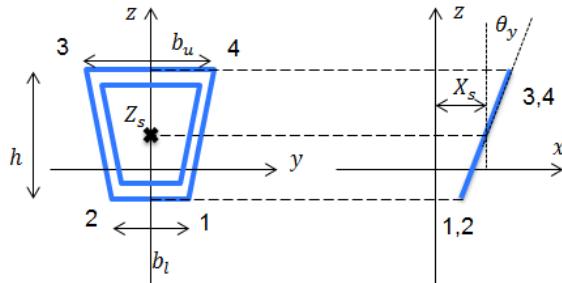


Figure 3.41: Rib geometric parameters for box component description

to real designs. Here generic connection and stiffening are considered on pylon, engine and wing sides. For this purpose the assembly of components described in figure 3.42 was defined. Each segment represents a cylinder, for instance the segment AB represent a cylinder with axis passing through A and B and basis in A and B respectively. Points A,B,C,E,F and G are on circle segments in planes orthogonal to  $x$  axis and centered on the engine middle axis. The cylinders AB,BC,EF and FG are on the engine, on the pylon box or on the wing and represent stiffeners, cylinders AG, FB, and EC represent links connecting the engine to the pylon or the pylon to the wing. Also this primitive needs to be symmetric with respect to the  $xz$  plane. For this reason F and B will lie on the  $xz$  plane and their  $x$  position and their  $z$  position also determines the distance of G and E and of A and C from the engine centerline.

It is possible to write all coordinates of the extremities:

$$A \equiv (x_1, -z_1 \sin(\theta_{x1}), z_1(1 - \cos(\theta_{x1}))) \quad (3.151)$$

$$B \equiv (x_1, 0, z_1) \quad (3.152)$$

$$C \equiv (x_1, z_1 \sin(\theta_{x1}), z_1(1 - \cos(\theta_{x1}))) \quad (3.153)$$

$$E \equiv (x_2, z_2 \sin(\theta_{x2}), z_2(1 - \cos(\theta_{x2}))) \quad (3.154)$$

$$F \equiv (x_2, 0, z_2) \quad (3.155)$$

$$G \equiv (x_2, -z_2 \sin(\theta_{x2}), z_2(1 - \cos(\theta_{x2}))) \quad (3.156)$$

The assembly being symmetric the radius of cylinders AB and BC are equal to  $r_1$ , the

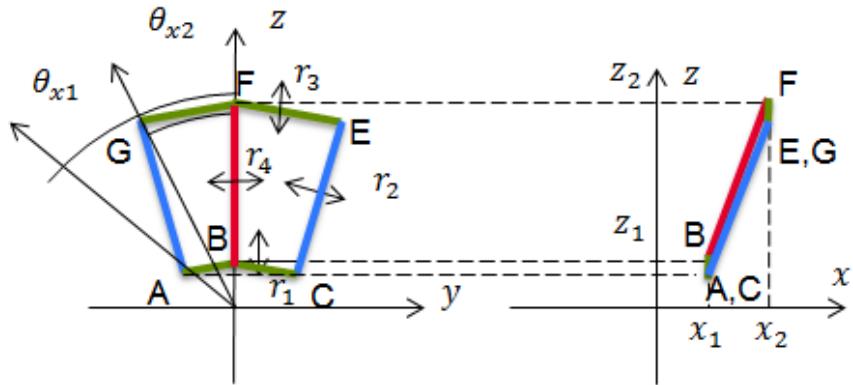


Figure 3.42: Engine and wing mount geometric primitive parametrization. Each line represents a cylinder. Points ABC and EGF lies on 2 conferences in planes orthogonal to the x axis and with center in the engine centerline.

radius of cylinders EC and GA are equal to  $r_2$ , the radius of cylinders EF and FG are equal to  $r_3$  and the radius of cylinder BF is  $r_4$ . For each mount, one has the vector of design variables  $\{x_t\} = \{x_1, z_1, x_2, z_2, \theta_{x1}, \theta_{x2}, r_1, r_2, r_3, r_4\}$ . With the knowledge of the connectivity of the solution, one can only enforce each configuration to be well connected defining explicit relationships of the form  $z_1 = z_1(x_1)$  and  $z_2 = z_1(x_2)$ . In this way the number of design variables is  $8n_t$  where  $n_t$  is the number of mounts assemblies.<sup>9</sup> The control points for convex hull constraints were only considered for the pylon box components in the position of box external vertexes. The final formulation for this problem is the same considered in equations 3.147, with the only change on the design variables meaning and on the lower and upper bounds. In figure 3.43 the initial configuration for the pylon and wing and engine mount topology optimization is represented using  $n_b = 2$  and  $n_t = 8$ . The design variables are then 64 for the engine and wing mounts and 26 for the pylon boxes. In table 3.6 some details for the optimization are given. The convergence history of Volume fraction, Stress aggregated constraints, TSFC variations and convex hull constraint is given in figure 3.44.

<sup>9</sup>One can observe that enforcing a connectivity between the engine, the pylon boxed and the wing mount, one could no longer speak of topology optimization. Actually the number of panels and assembly in the final solution evolves varying also the load path. In this case one could say that this optimization is somewhere between a shape and a topology optimization.

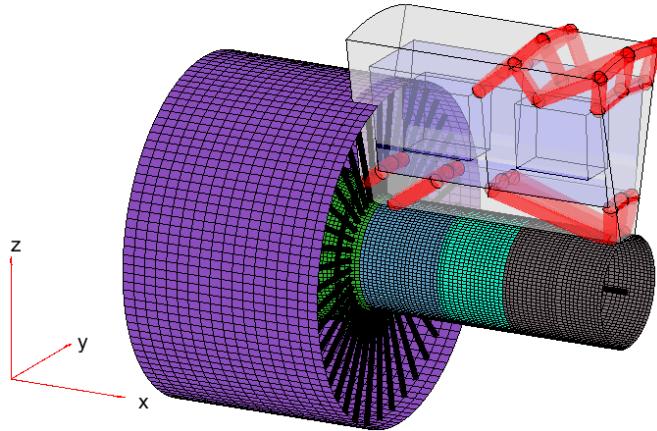


Figure 3.43: Initial point configuration of the topology optimization including 2 pylon boxes and 4 engine mounts and 4 wing mounts cylinder assemblies.

The final volume fraction is higher than the one found in the optimal configuration of SIMP and using MNA with rectangular parallelepiped components. This is probably due to an excessive restriction of the design domain. A possible conclusion could be that, in this case making a pylon using planar panels assembled to make pylon boxes and connecting them using cylinder assemblies is not as effective as building a truss pylon. Stress constraints are slightly violated at convergence but this is actually due to MMA behavior. In fact it can happen that during convergence history the optimization step becomes too big and MMA starts violating constraints. On the other hand the stress violation is small enough, 0.02 % and can be considered as acceptable. All other constraints are satisfied. In particular for this case the Convex hull constraint is not even activated at convergence. The component plot and the density plot of the solution are reported in figure 3.45 and 3.46. In the latter only half of the solution is represented for clarity. One can observe that in the frontal pylon box, 3 panels vanished, letting only lateral panels. From figure 3.46 one can observe that the solution presents some large gray region. For this reason it is important to make a verification of the final solution on a thresholded solution as it was the case for SIMP solutions. Using a threshold of 0.45 the final configuration has the performances shown in table 3.7. The volume fraction was increased from 8.88% to 9.79%, the final aggregated KS function is negative and the value of the aggregated TSFC variation is reduced to 0.1698. To avoid such modification a continuation strategy on the value of  $g_t$  could be adopted but is left as possibility for future works. The displacement and the von Mises stress plots are shown in figure 3.47. It is possible to recognize again that the frontal part composed by the wing mounts and the lateral panel

of the frontal box principally works under lateral and vertical loads as the rear structure activates under the axial load on the engine. This seems to confirm a principle of design that is valuable for all results reported in this thesis for the different approaches adopted to solve topology optimization with 3 load cases and with both stress and performance constraints.

Table 3.6: Optimization problem set up. The hypothesis made to get numerical results for the 3D MNA topology optimization using pylon boxes and cylinder assemblies on the multiple load case are listed here

Symbol	Name	Value
$E_0$	Young's Modulus	210 GPa
$\nu$	Poisson Ratio	0.29
$N_{el}$	Number of elements in the design zone	486400
$N_{DOFs}$	Number of rows of the stiffness matrix	1529847
$N_G$	Number of Gauss points in the design zone	3891200
$N_{var}$	Number of Design variables	90
$p_b$	MNA penalty	3
$g_t$	MNA gray thickness	2 × average mesh size
$\kappa$	IE geometric assembly aggregation constant	10
$P$	Aggregation constant for the stress constraint	4
$P_l$	Aggregation constant for load case	10
$\sigma_{lim}$	Allowable stress to be used in (2.58)	10 MPa
$\sigma_{alw}$	Maximum local allowable stress	49 MPa
$T_0$	Allowable consumption variation	0.17 %
$\{\rho f\}$	Inertial load	{0}
	Stopping condition on the Karush-Kuhn Tucker residual norm	$KKTn \leq 10^{-3}$
	Stopping condition on the iteration number	$iter \geq 300$
	MMA external move limit for first 2 iterations	0.4
	MMA maximum asymptote distance from the current point	0.1

Table 3.7: Solution responses before and after thresholding

Response	Original solution	After thresholding ( $t_{sh} = 0.45$ )
$\Delta TSFC\%$	0.17	0.1698
$V\%$	8.88	9.79
$G_{KS}^l$	$2.22 \times 10^{-2}$	$-2.2 \times 10^{-2}$
$\max(\sigma_{VM})$	37.09 MPa	30.08 MPa

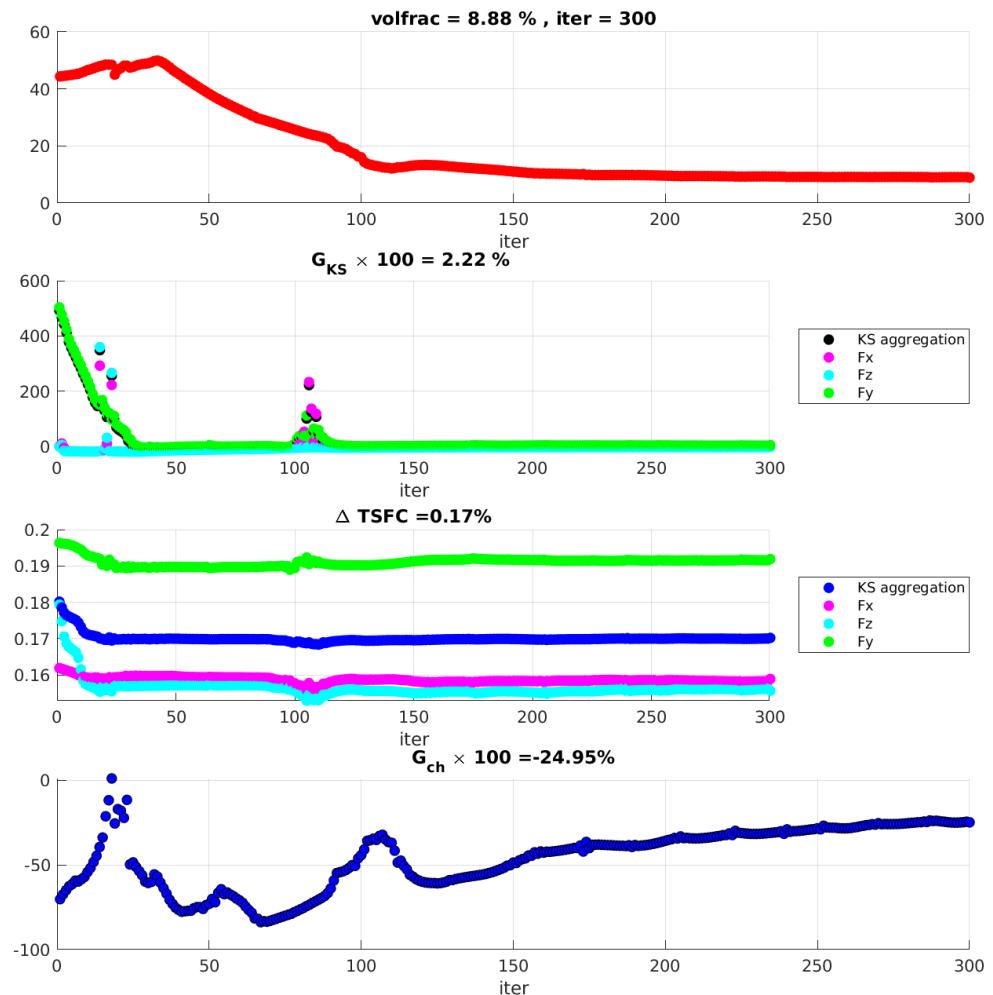


Figure 3.44: Convergence history of the multi load case topology optimization using MNA with pylon boxes and cylinder assemblies primitives

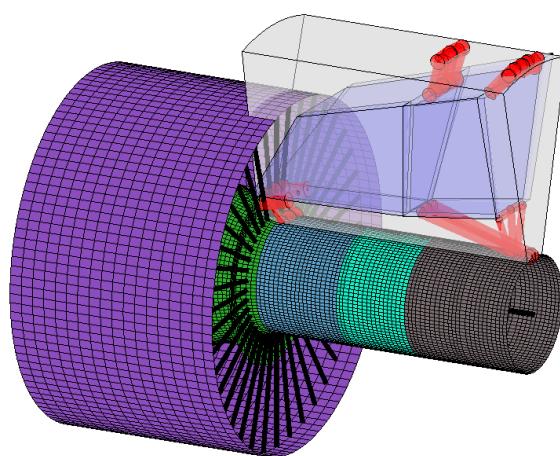


Figure 3.45: Component plot of the solution of MNA with pylon boxes and cylinder assemblies primitives.

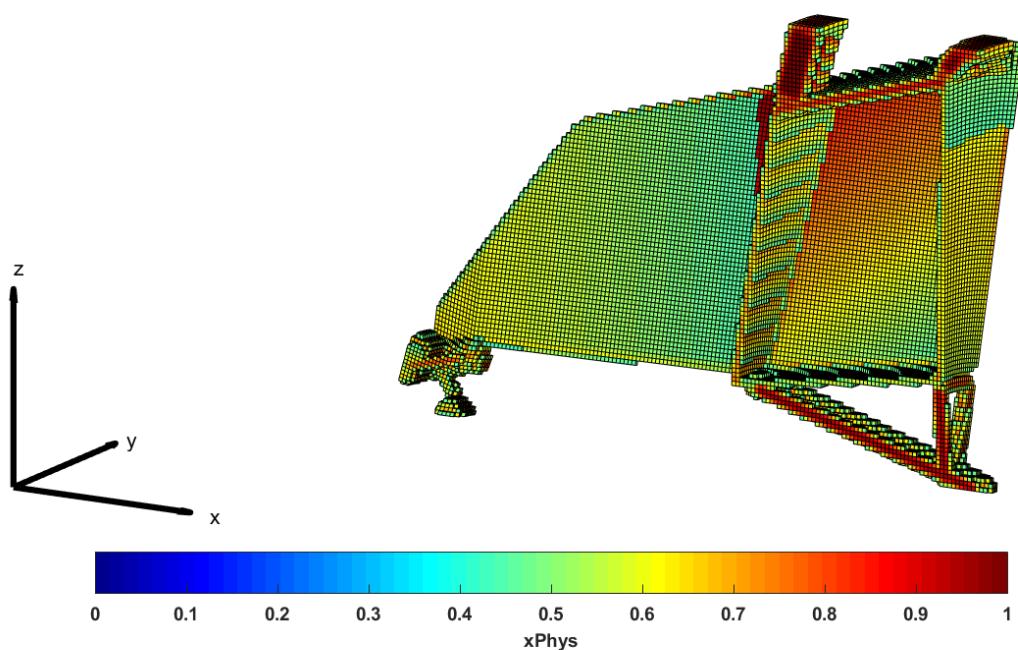


Figure 3.46: Density plot of half solution ( $y \geq 0$ ) of the multi load case analysis for the pylon and engine mount design using MNA 3D with pylon boxes and cylinder assemblies primitives. Only elements with a physical density greater than 0.45 are represented.

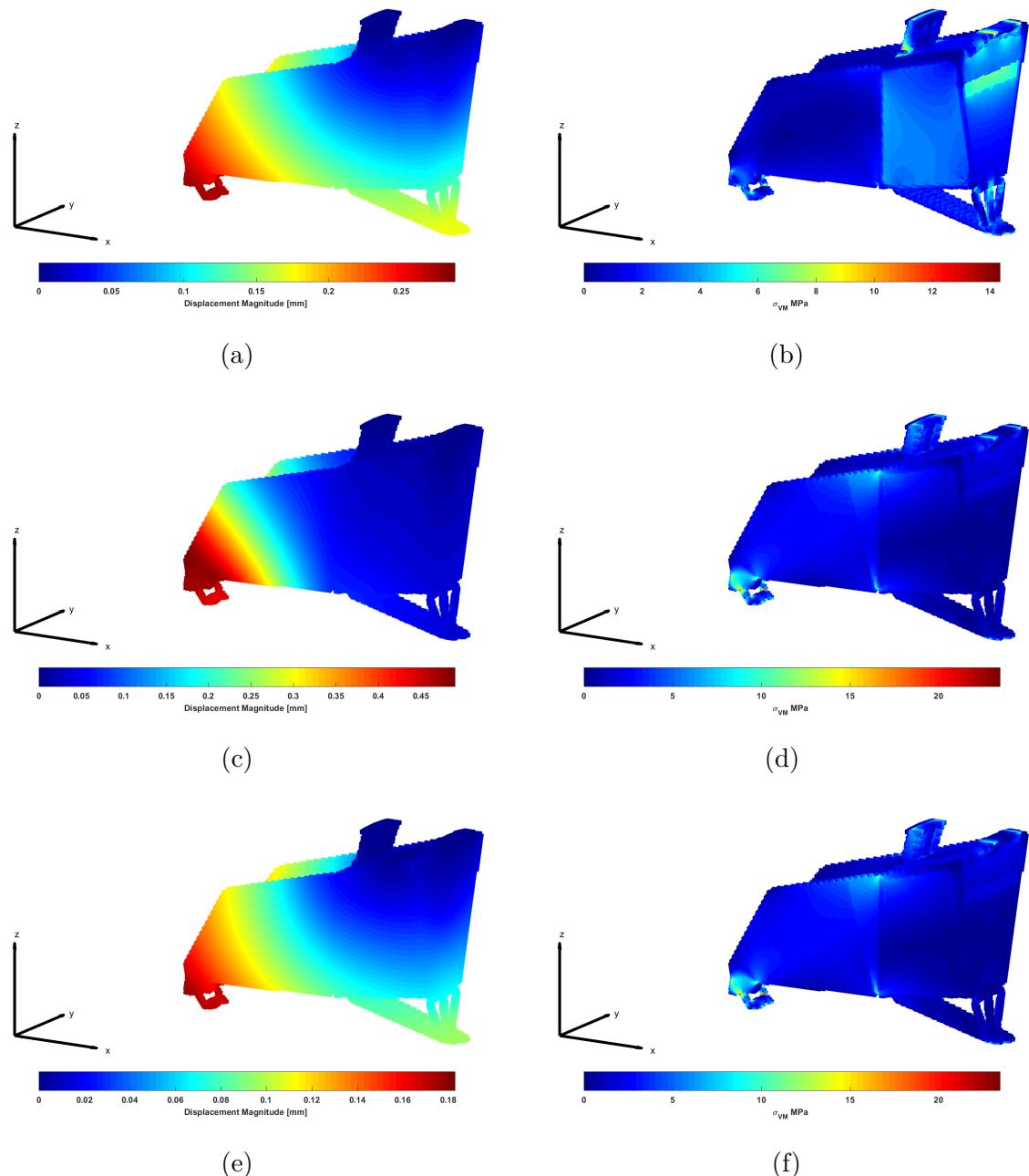


Figure 3.47: Topology optimization results for MNA with pylon boxes and cylinder assemblies primitives. Displacement magnitude color map (a) for  $F_x$ , (c) for  $F_y$ , (e) for  $F_z$  respectively. von Mises stress color map (b) for  $F_x$ , (d) for  $F_y$ , (f) for  $F_z$  respectively.

### 3.11 Summary and conclusions

In this chapter the Lagrangian approaches to topology optimization were considered. In the proposed framework this family of approaches was needed for exploration and benchmarking. Therefore a literature review is proposed to cover Moving Morphable Components, Geometry Projection and Moving Node Approach. A novel theoretical framework called Generalized Geometry Projection (GGP), is then proposed to cover all the reviewed techniques. Not only does this technique provides a unique implementation workflow for the reviewed strategies but also provides an efficient way to remove saddle points from model responses in the optimization formulation. This was proven in a 2D cantilever use case that was analyzed for different methods and number of Gauss points in each sampling window of GGP. To use efficient gradient based solvers the adjoint evaluation of sensitivities is provided. Stress based constraints are implemented in 2D with unified aggregation and relaxation approach and an update strategy of allowable was considered to control the actual maximum von Mises stress in the solution. Finally the MNA approach was extended to solve the design of pylon and engine mount with stress and consumption constraints under multiple load case. To this end 3D components specific to the pylon, engine and wing mounts design were proposed and tested. To summarize, this chapter covered the following topics:

- Topology optimization using Moving Morphable Components with ersatz material model.
- Topology optimization using Geometry Projection
- Topology optimization using Moving Node Approach.
- The geometric assembly
- Stress constraint in topology optimization using explicit frameworks
- 3D extensions
- Application of a Lagrangian approach to the engine to wing attachment design problem.

The novel contributions introduced in this chapter consist in:

- The definition of a novel strategy that called Generalized Geometry Projection that can recover existing strategies.
- The definition of an improved geometric assembly strategy called saturation that helps in avoiding common techniques' shortcomings.
- The implementation of a technique similar to the one of [Le et al., 2010] also helps enforcing maximum constraint adapting the allowable value during iterations.
- The definition of a library of simple components that can be treated using signed distance and geometric operations on activation functions.

- The implementation of a convex hull constraint that enforces components to lie inside the convex hull of the design region.
- The application of MNA to the pylon and engine mount topology optimization problem.
- The application of pylon like components using combinations of the aforementioned library of simple components to the pylon and engine mount topology optimization problem.
- This work led to the publication of a journal paper [[Coniglio et al., 2019b](#)].

As conclusions here it is possible to list that:

- It is possible to define a unified approach that can represent existing techniques in a unique mathematical framework. Such an approach also overcomes some difficulties linked with all Lagrangian approaches. Some practical recommendations were given to avoid saddle points in the optimization problem.
- Making the union of components is a common task of an explicit framework that can introduce some unwanted effects in the FEM used to evaluate a geometric configuration. The proposed saturation strategy can avoid intermediate densities in regions where only one component is activated.
- Enforcing black and white solutions within an explicit framework is possible, adopting a continuation strategy on the transition zone width. The maximum stress can be controlled thanks to the proposed adaptive strategy. On the other hand, the solutions found could include stress singularity that are not well represented by the finite element model. This is also due to the fact that is due to the fact that using relaxation one can find singular optima, but this doesn't remove the other local optima from the feasible design domain.
- The migration from SIMP to MNA framework is straightforward. This kind of framework allows an improved mechanical understanding of the solution, a faster post processing and the introduction of manufacturing considerations. Enforcing that a solution is only composed by boxes and cylinders is for example useful to ensure that the solution is actually manufacturable. Since not all solutions representable with SIMP approach can be represented with Lagrangian approaches, sometimes MNA solutions are suboptimal with respect to SIMP solutions. This problem still is in large part dependent on the number of components considered in the optimization problem and on their shapes.
- The solutions obtained with the 3D framework are consistent with results found in Chapter 2. The load path is always basically the same found for all 3 formulations. The differences lie in some structural details that may or not be described by each formulation. Reducing the space of possible solutions the performances are impacted.

Still some shortcomings need to be addressed and were not studied in this thesis. The GGP proposed strategy could actually be improved by sensitivity modifications as it was explained

in subsection 3.8.3. Moreover the same limitations studied in chapter 2 concerning the model are also valid for this chapter and should be considered as research axes to improve the presented framework. The projection approaches also opens new perspectives for both the implementation of advanced manufacturing considerations and for the simplification of the optimization problem. These should be investigated further in future works. Optimization algorithms employed in this thesis were local. In this context it would be possible to consider a progressive improvement of local minima as in [Zhang and Norato, 2018].

# Conclusions and perspectives

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The goal of this work consists in developing a methodology for the quantification of the impact of engine integration design on tip-clearance variations (related to fuel consumption) and for the exploration of innovative designs. Topology optimization is identified as the most suitable technique to deal with such a problem without making too strong assumptions on the final design. Therefore, a topology optimization framework compatible with industrial applications is proposed. The main contributions of this thesis can be divided in three main topics that reflect the structure of this thesis. A first chapter deals with numerical challenges concerning the use of finite element analysis for the evaluation of tip-clearance variations and von Mises failure criteria. The second chapter shows how to include such analyses inside a classical topology optimization (SIMP) Eulerian framework. Finally, the third chapter demonstrates the use of Lagrangian approaches in topology optimization for both compliance and stress-based formulations. All the 7 numerical challenges that were identified in the introduction are finally addressed:

### 1. Dealing with industrial models

The proposed topology optimization framework is compatible with Abaqus or Nastran engine finite element models. The technique adopted for this purpose is superelement formulation which reduces the simulation and optimization computational burden but keeping the same accuracy of a simulation including the whole engine model. Tip clearance post processing is also shown to be compatible with this technique.

### 2. Dealing with complex design space geometries

The proposed finite element framework uses 8 node brick elements with full integration. These can form structured meshes that can easily accommodate complex design zone geometries.

### 3. Dealing with non-consistent meshes

The design zone mesh and the engine mesh are not constrained to be coincident at their interface. This result is achieved through the implementation of RL-RBF, Internodes and Mortar techniques in the proposed framework. In the author's contribution [Coniglio et al., 2018a] two new strategies were proposed: the so-called Weighted Average Continuity Approach and moment correction. The first achieves a tradeoff between computational burden, algorithmic complexity and accuracy while the second enforces the balance of mechanical momentum of all reviewed strategies even for curved interfaces.

#### 4. Efficiency and scalability

The use of iterative approaches with different multigrid preconditioners and smoothers is implemented. Several smoothers are compared for the engine use case. An investigation on tuning parameters is conducted to improve the solver efficiency. To avoid the selection of a new optimal value per iteration a Line-Search strategy is proposed and compared with the result of classic approaches. All tested methods are compatible with distributed memory environments.

#### 5. Employing Lagrangian Approaches

MMC, MNA and GP strategies are available in the literature. In the author's contribution it was also shown that these formulations are particular cases of a unified formulation that was named Generalized Geometry Projection [Coniglio et al., 2019b]. The relationships between formulations' parameters and optimization related issues like the presence of saddle points in the optimization problem were also studied and practical solutions were provided. The proposed continuation strategy for stress-based topology with MNA approach can be used to achieve improved optima with small area of intermediate densities.

#### 6. Dealing with stress constraints

Stress constraints can be considered during the optimization thanks to the Unified Aggregation Relaxation approach. This helps achieving singular optima and reduces the computational burden associated with local stress constraints. Moreover, this strategy was combined with the MNA approach as a particular case of GGP, which also constitutes a novel contribution of this work.

#### 7. Employing specific geometric primitives

The use of pylon like components that can change shape and number of panels/bars is rendered possible within the proposed GGP approach. In this way the solution is enforced to be compatible with standard manufacturing techniques.

The proposed methodology is tested on a demonstration engine model [Coniglio et al., 2019a]. It was demonstrated that it is possible to include tip clearance variation criteria in the topology optimization loop, even in the very preliminary design phases. The only requirement for such analysis is the availability of a Whole Engine Model capable of post processing tip clearance variations. Topology optimization using SIMP, MNA with rectangular parallelepiped components and MNA with pylon box components approaches provide different solutions. But all these solutions can be considered as variants of the same architecture (exhibiting the same load path). By making a parametric study on the value of the allowable fuel consumption variation it is also possible, using the SIMP approach, to identify a trend in the solution load path that can be used as design driver for integration engineers. This work led to the publication of three journal papers [Coniglio et al., 2018a, Coniglio et al., 2019a, Coniglio et al., 2019b] and to two conference articles [Coniglio et al., 2017, Coniglio et al., 2018b].

## 4.1 Perspectives

The proposed approach can be used to solve several engineering problems in which displacements control is required on a linear industrial model. The proposed library of geometric primitives can be enriched to treat different industrial problems. Still many developments could improve the proposed framework. The extension of the proposed framework to unstructured finite element mesh could give even more freedom to the shapes that can be considered for the design space. Distributed Memory environments and GPU accelerators can be used to further refine the solution. This would also make it possible to consider larger design space and include several other subcomponents in the optimization, such as nacelle, air inlet and pylon secondary structures. Optimization algorithms employed in this thesis were all local. In this context it would be possible to consider improvement techniques such as tunneling [Zhang and Norato, 2018] aimed at avoiding that the algorithm gets stuck in local optima. Surrogate models could also be employed to improve the overall optimization efficiency especially for very expensive models using Efficient Global Optimization (EGO) [Bouhlel et al., 2018]. This could be effective especially for Lagrangian Approaches when the number of design variables can be reduced to less than 100. Different formulations of stress relaxation such as the one considered in [Zhang et al., 2017a] should be tested, to try to avoid convergence to nonsingular optima in general. Other important physical analyses could be integrated in this framework such as dynamic analysis, non-linearity, thermo-mechanical analysis, aerodynamic or multi-physic analysis. For example, an application of such a framework including 2D nonlinear finite element model was investigated in a recent contribution for the design of a compliant mechanism of a morphing wing [Capasso et al., 2019]. The main challenge that needs to be addressed to consider such simulations, is the fact that superelements, that require a linear static engine model to be employed, should be replaced by other techniques such as non-intrusive coupling [Gendre et al., 2009]. Another major issue consists in the use of tying relationships at the interfaces. Engines being mounted under pylons and being subject to large temperature variations, need to be connected by a nearly isostatic system of engine mounts. Therefore, the solution proposed by the proposed framework could need some changes in the kinematic link that they have with the engine and this could deprecate their performances. Future works could address these issues to enforce more reliable results typically including the maximization of compliance induced by temperature variations load cases in the optimization formulation. Another interesting challenge for future work consists in making a topology optimization consistent with fatigue requirements in large structures. In fact, a common assumption of stress-based topology optimization is that allowable stress is known *a priori* in each structural component and is not influenced by the solution shape or by the manufacturing process. Actually, in large structures, the number of parts used to make the final assembly is unknown *a priori*, the technology adopted for the manufacturing process depends on the shape of the solution, the allowable are also local properties of the solution. On top of that this hypothesis also limits the application of technologies like 3D printing to structures where fatigue considerations are important. In fact, 3D printed structures can have large range of fatigue allowable stress that depends on the thermal history experienced by the material in each point, on the residual stresses and on the final surface roughness. These aspects depend on the manufacturing process that is a consequence of the

design shape. Indeed, the link between design geometry and manufacturing process is not only essential to ensure that a design can be manufactured but also to efficiently design a 3D printed structure. This kind of correlation between material properties and manufacturing process is also valid for other technologies such as composite structures. An approach that ensures the existence of a manufacturing process thanks to geometric primitives that rely on 3D printing manufacturing process was proposed and implemented in [Bhat et al., 2019]. The main idea is to determine a design as the consequence of a manufacturing process. Therefore, optimization design variables can be directly those of the manufacturing process instead of geometry parameters. In the future such approach could be used to simulate manufacturing process to get real material properties such as fatigue allowable stress. Then these properties could be projected on the finite element model simulating the component operating conditions. In this way the design optimization could take in account the effect of manufacturing process on final allowable stress. This would then ensure more realistic fatigue life estimation for 3D printed designs.

## APPENDIX A

# Appendix 1

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This appendix reports the detailed error analysis of each configuration benchmark of section 1.3.10.

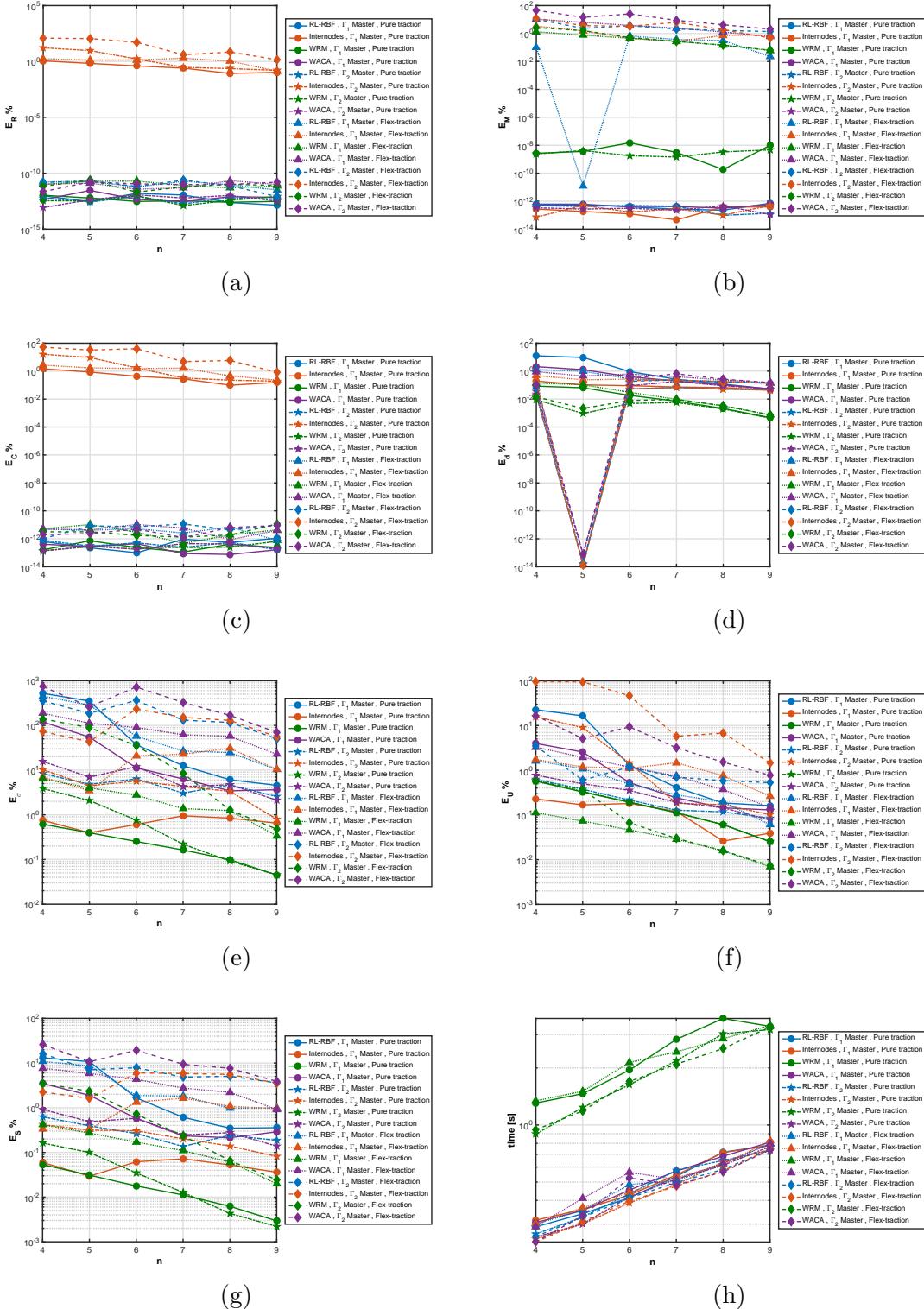


Figure A.1: Benchmark results in configuration (1) impact of  $n$  over: (a)  $E_R$ , the Resultant Force relative error, (b)  $E_M$ , the moment relative error, (c)  $E_c$ , the interface compliance relative error, (d)  $E_d$ , the displacement discontinuity relative error, (e)  $E_\sigma$ , the maximum of Von Mises stress relative error, (f)  $E_U$ , interface displacement field relative error, (g)  $E_S$ , average Von Mises stress relative error, (h) CPU time (s).

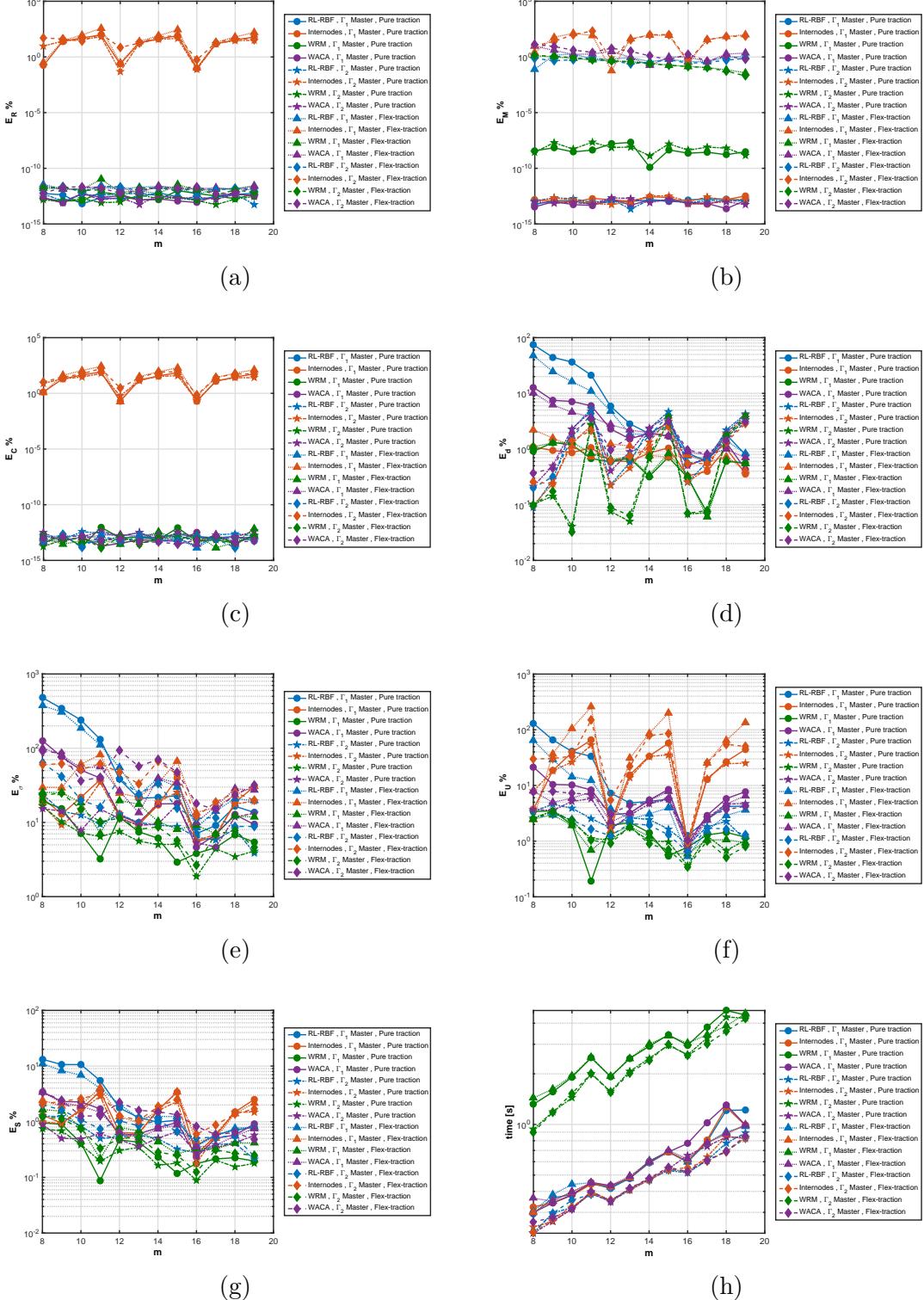


Figure A.2: Benchmark results in configuration (2) impact of  $m$  over: (a)  $E_R$ , the Resultant Force relative error, (b)  $E_M$ , the moment relative error, (c)  $E_c$ , the interface compliance relative error, (d)  $E_d$ , the displacement discontinuity relative error, (e)  $E_\sigma$ , the maximum of Von Mises stress relative error, (f)  $E_U$ , interface displacement field relative error, (g)  $E_S$ , average Von Mises stress relative error, (h) CPU time (s).

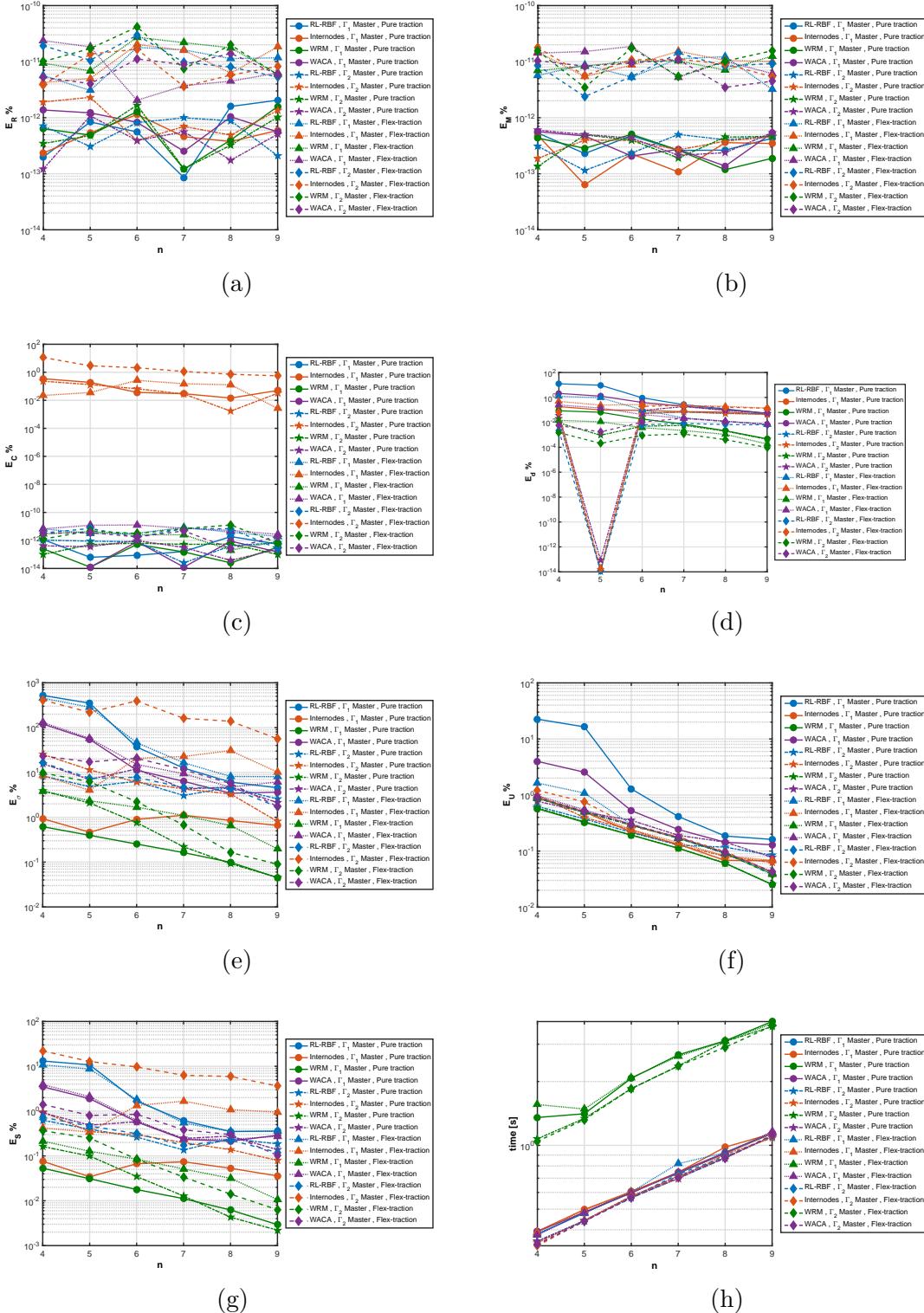


Figure A.3: Benchmark results in configuration (1) after moments balance correction. Impact of  $n$  over: (a)  $E_R$ , the Resultant Force relative error, (b)  $E_M$ , the moment relative error, (c)  $E_c$ , the interface compliance relative error, (d)  $E_d$ , the displacement discontinuity relative error, (e)  $E_\sigma$ , the maximum of Von Mises stress relative error, (f)  $E_U$ , interface displacement field relative error, (g)  $E_S$ , average Von Mises stress relative error, (h) CPU time (s).

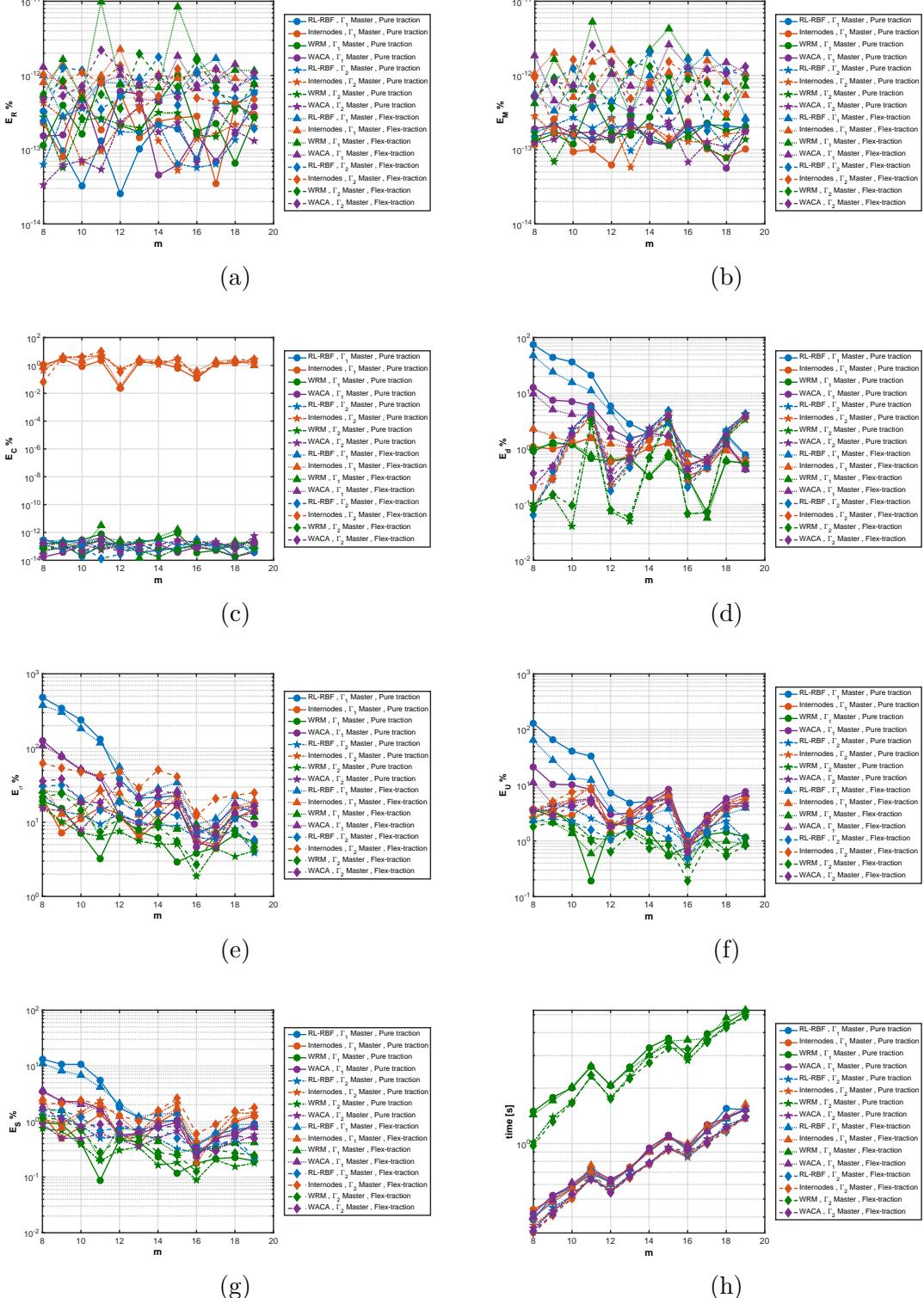


Figure A.4: Benchmark results in configuration (2) after moments balance correction. Impact of  $m$  over: (a)  $E_R$ , the Resultant Force relative error, (b)  $E_M$ , the moment relative error, (c)  $E_c$ , the interface compliance relative error, (d)  $E_d$ , the displacement discontinuity relative error, (e)  $E_\sigma$ , the maximum of Von Mises stress relative error, (f)  $E_U$ , interface displacement field relative error, (g)  $E_S$ , average Von Mises stress relative error, (h) CPU time (s).



## APPENDIX B

# Appendix 2

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## B.1 2D topology optimization examples

### B.1.0.1 MBB optimization

The MBB beam optimization problem is a very common test case in topology optimization. It is often modeled considering the symmetry of geometry, loads and boundary conditions c.f. figure B.1. Two meshes were adopted to solve this problem one of  $150 \times 50$  and the other of

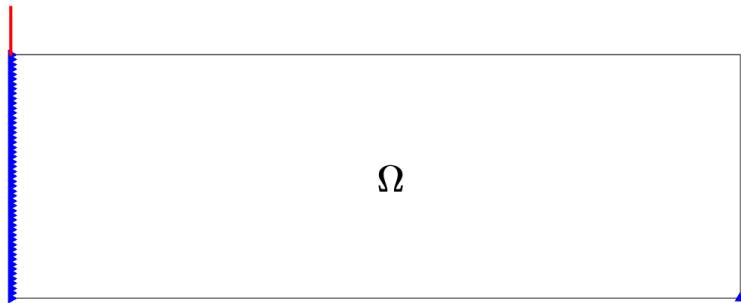


Figure B.1: Geometry, Load and boundary conditions of the MBB topology optimization problem. Blue triangles are oriented as the fixed DOFs. The red arrow represents the applied load and point through the position of node were the load is applied.

$300 \times 100$ . To have a mesh independent solution the initial radius was set in the first case to 6 and in the second to 12. The results and the convergence history of each optimization is shown in figure B.2. One can observe from figure B.2c and B.2a that the solution found is

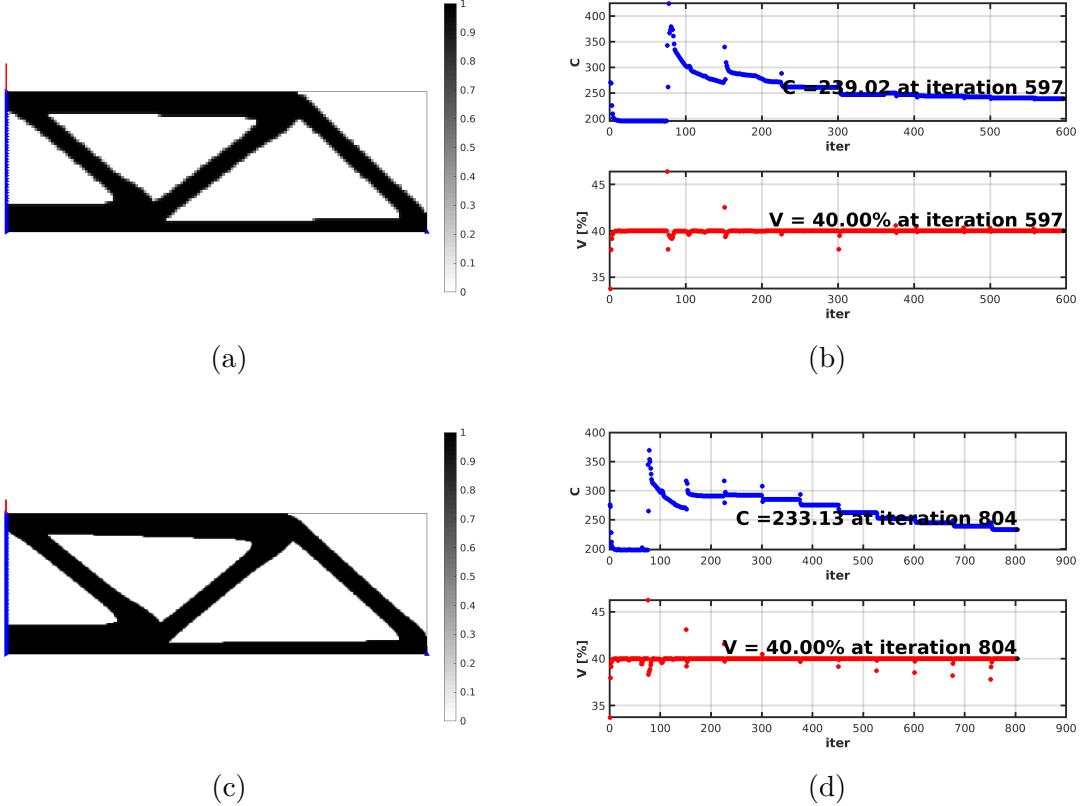


Figure B.2: MBB topology optimization results. (a)-(b)  $\{x_{Phys}\}$  distribution and responses convergence for the  $150 \times 50$  discretization.(c)-(d)  $\{x_{Phys}\}$  distribution and responses convergence for the  $300 \times 100$  discretization

independent from the mesh found. The choice of both filer radius and continuation strategy helps in the determination of this design. Moreover the solution have small transitions region on the boundary of the solid. This is thanks to the Heaviside filter and the continuation strategy adopted for  $\beta$ . Thanks to that, exploiting this results is easier. In fact using a threshold it is possible to convert these solutions to a black and white (0-1) design that will be close enough to the actual optimization solution. On figures B.2d and B.2b one can observe a quite long converge history for both meshes. This is due to the continuation strategy adopted here. Moving forward, this design is an actual local minimum that respect all constraints as it can be observed from both figures B.2d and B.2b. In fact even if the convergence criterion on design vector change was not reached, in the final iterations the variations of both objective function and volume constraint are very small. The FEM is still valid on the optimum even if  $150 \times 50$  is slightly softer. This can be an effect of differences in the solutions. The small displacement hypothesis cannot be verified here since all measures considered for this test case are dimensionless. Finally being the final application and the technology adopted for the manufacturing not specified, it is not possible to make any conclusions about the acceptability of this design candidate or of the model validity.

### B.1.0.2 Short cantilever beam optimization

This second use-case is also very well known and studied in the literature. A domain design is clamped on the left side and excited on the right side c.f. figure B.3. Two meshes were

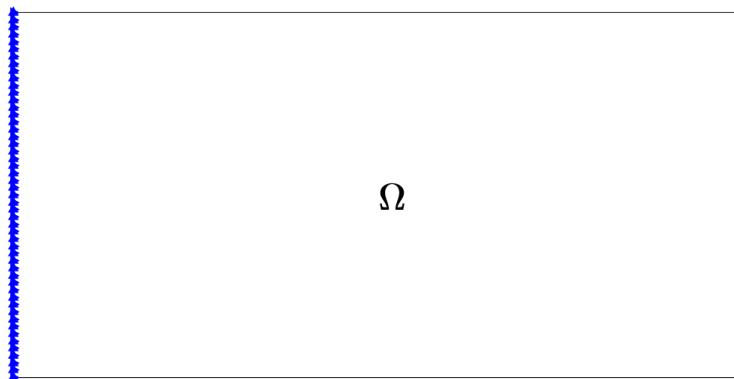


Figure B.3: Geometry, Load and boundary conditions of the short cantilever topology optimization problem. Blue triangles are oriented as the fixed DOFs. The red arrow represents the applied load and point through the position of node were the load is applied.

adopted to solve this problem one of  $100 \times 50$  and the other of  $200 \times 100$ . To have a mesh independent solution the initial radius was set in the first case to 6 and in the second to 8. The results and the convergence history of each optimization is shown in figure B.4. The solutions obtained for different discretizations are also this time very similar c.f. figures B.4a and B.4c. Nearly black and white solutions are determined. The convergence figures history B.4b and B.4d, tell us that both optimization were converged this time respecting the design variable change criterion. Both solution found are symmetric with respect to the middle axis. Also this time one can identify 8 thin members composing the solution. The design found by the finer discretization is slightly stiffer than the one found by the coarser mesh. Also this time one can find an explanation either in small discrepancies between solutions. Again any conclusions can be outlined on the model validity and the design acceptability on the base of the data of the problem in hand.

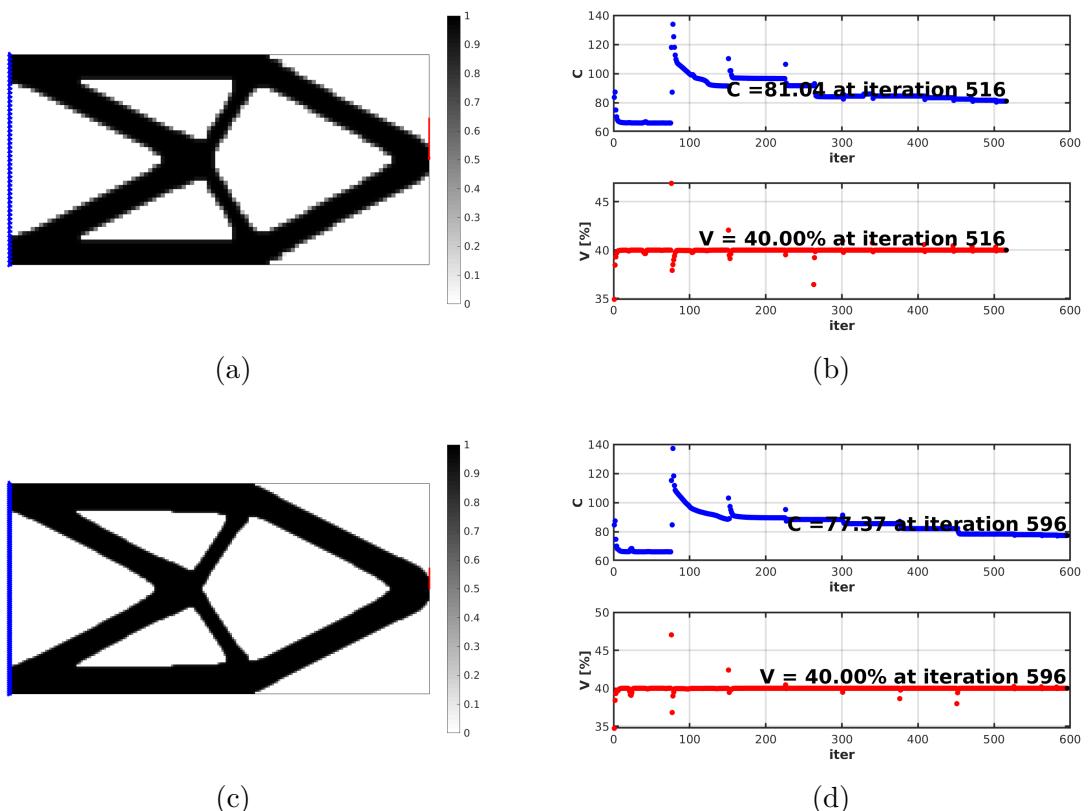


Figure B.4: Short cantilever topology optimization results. (a)-(b)  $\{x_{Phys}\}$  distribution and responses convergence for the  $100 \times 50$  discretization.(c)-(d)  $\{x_{Phys}\}$  distribution and responses convergence for the  $200 \times 100$  discretization

## B.2 KS function properties

The following subsection reviews essential properties of the Kreisselmeier-Steinhaus (KS) functions used for the stress constraints. Given that  $P(\bar{g}_i - \bar{g}_{max}) \leq 0, \forall i, e^{P(\bar{g}_i - \bar{g}_{max})} \leq 1, \forall i$  and  $\frac{1}{P} \ln \left( \sum_{i=1}^{N_G} e^{P(\bar{g}_i - \bar{g}_{max})} \right) - \frac{\ln(N_G)}{P} \leq 0$  so that one can conclude  $G_{KS}^l \leq \bar{g}_{max}$  and knowing that  $\sum_{i=1}^{N_G} e^{P(\bar{g}_i - \bar{g}_{max})} > 1$  implies  $\frac{1}{P} \ln \left( \sum_{i=1}^{N_G} e^{P(\bar{g}_i - \bar{g}_{max})} \right) > 0$  so that:

$$\bar{g}_{max} - G_{KS}^l = \frac{\ln(N_G)}{P} - \frac{1}{P} \ln \left( \sum_{i=1}^{N_G} e^{P(\bar{g}_i - \bar{g}_{max})} \right) < \frac{\ln(N_G)}{P} \quad (\text{B.1})$$

As a remarkable results:

$$G_{KS}^l \leq \bar{g}_{max} < G_{KS}^l + \frac{\ln(N_G)}{P} = G_{KS} \quad (\text{B.2})$$

Where  $G_{KS}$  is the Kreisselmeier-Steinhaus function. Equivalently:

$$0 \leq \bar{g}_{max} - G_{KS}^l < \frac{\ln(N_G)}{P} \quad (\text{B.3})$$

Finally  $\lim_{P \rightarrow \infty} \frac{\ln(N_G)}{P} = 0$  that for eq. B.3 implies:

$$\lim_{P \rightarrow \infty} G_{KS}^l = \bar{g}_{max} \quad (\text{B.4})$$

One can also limit the range of variation of  $\bar{g}_{max}$  using an allowable tolerance:

$$\left| G_{KS}^l - \bar{g}_{max} \right| < \epsilon \quad (\text{B.5})$$

According to eq. (B.1) one can chose a value of P of

$$P > \frac{\ln(N_G)}{\epsilon} \quad (\text{B.6})$$

One must also keep in mind that for greater value of  $P$ , the  $G_{KS}^l$  can have very non-linear behavior. This has some negative consequences on gradient based optimization solver. An opposite strategy to deal with stress non linearity was explored in Lian et al. [Lian et al., 2017]. In this paper they proposed to increase the number of Gauss Points  $N_G$ . According to equation (B.1) this strategy increase the distance from the true maximum function hence reducing the stress aggregation function non linearities. When using lower bound KS function and p-mean this difference is inconvenient because the final maximum stress will be greater than the one corresponding to KS aggregation, needing compensation on the stress limit (as suggested in this paper). On the other hand using KS function or p-norm the design is always slightly oversized, so that increasing the aggregation constant could still be beneficial to the final mass of the solution. A less elegant, but still effective way of accounting for this gap is to change allowable stress during iterations in the way proposed in Le et al. [Le et al., 2010].

### B.3 TSFC constraint parametric study

This section proposes to study the impact of the TSFC variation constraint on the solution of problem in Eq. 2.100. This is interesting to extract some conclusions on:

- The quantification of the possible impact of pylon and engine mount design on the TSFC variation.
- The quantification of the mass investments required to achieve a given tip clearance control improvement.
- The detection of key design features that can help controlling tip clearance.

Solution A, B, C, D represented in figures B.6,B.7,B.8,B.9 are obtained with  $T_0 = 0.16, 0.17, 0.18, 1$  respectively. The corresponding value of aggregated TSFC variation and aggregated Compliance is shown figure B.5. One can then outline the following conclusions:

- When lower value of TSFC variation are enforced, the final volume fraction is increased, that shows that these two responses are antagonistic.
- The solution D has a final aggregated TSFC variation of 0.183 smaller than the bound of 1. This means that TSFC constraint is not active for this optimization and that the only stress constraint was activated.
- Solution C and B improve both compliance and TSFC variation compared to solution D. On the other hand solution A is heavier and softer than the other solutions.
- Solution A (c.f. figure B.6) is very different from the other solutions (c.f. figures B.7,B.8 and B.9). The frontal part is stiffened but has a much softer rear part with respect to lateral loads.
- The frontal part is stiffened passing from solution D to solution A, in particular connection between core casing and Outlet Guide Vain basis is reinforced. This seems to be a key connection for engine deformation control.

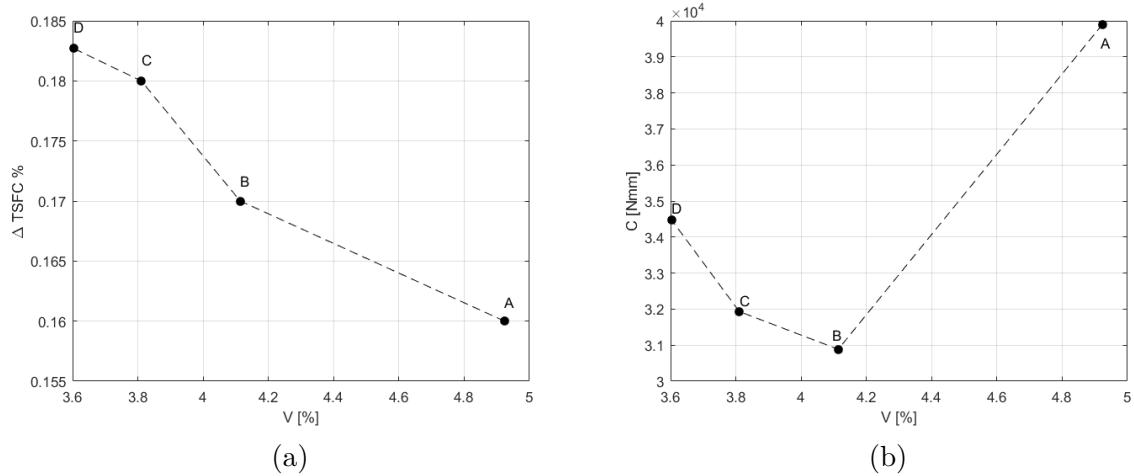


Figure B.5: Pareto front obtained varying the allowable aggregated TSFC variation ( $T_0$ ) in equation 2.100. a)  $V$  vs  $\Delta TSFC \%$  corresponding to each solution. b)  $V$  vs  $C$  % corresponding to each solution.

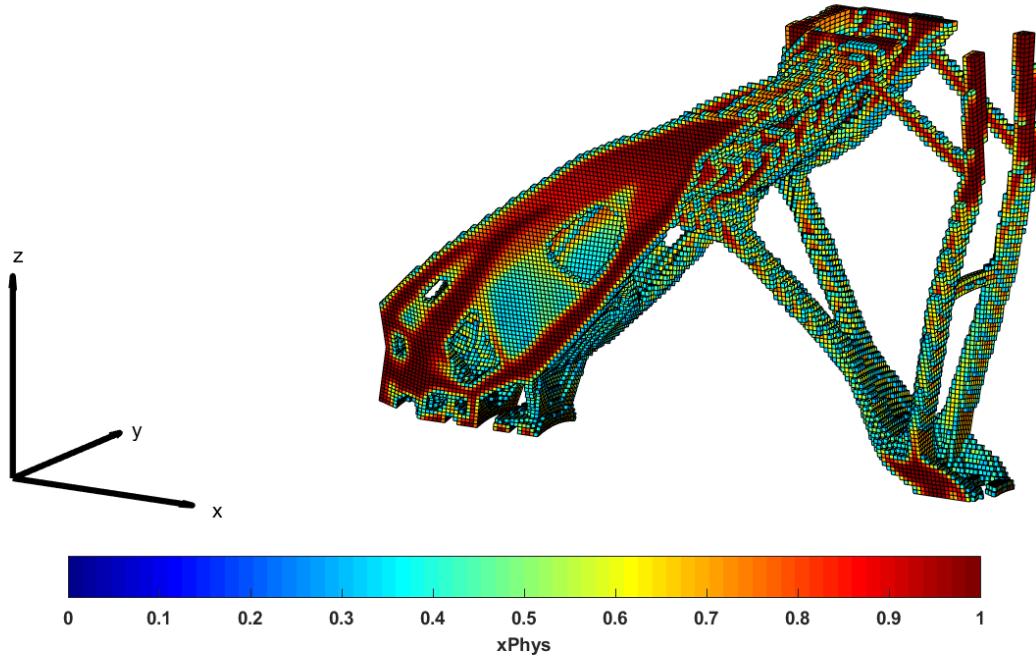


Figure B.6: Density plot of solution A obtained with  $T_0 = 0.16$

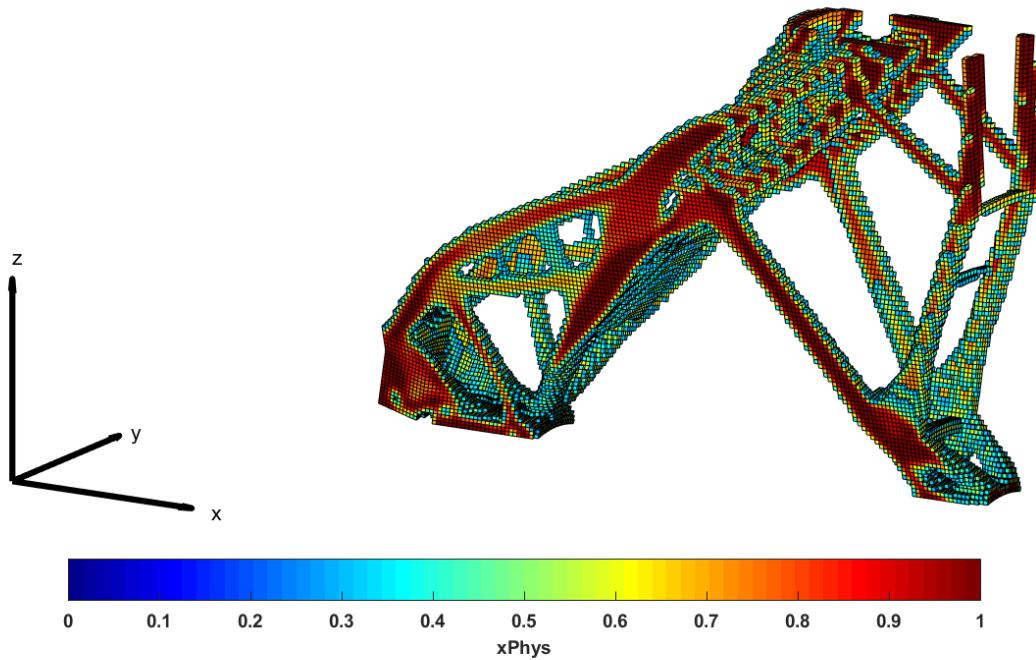


Figure B.7: Density plot of solution B obtained with  $T_0 = 0.17$

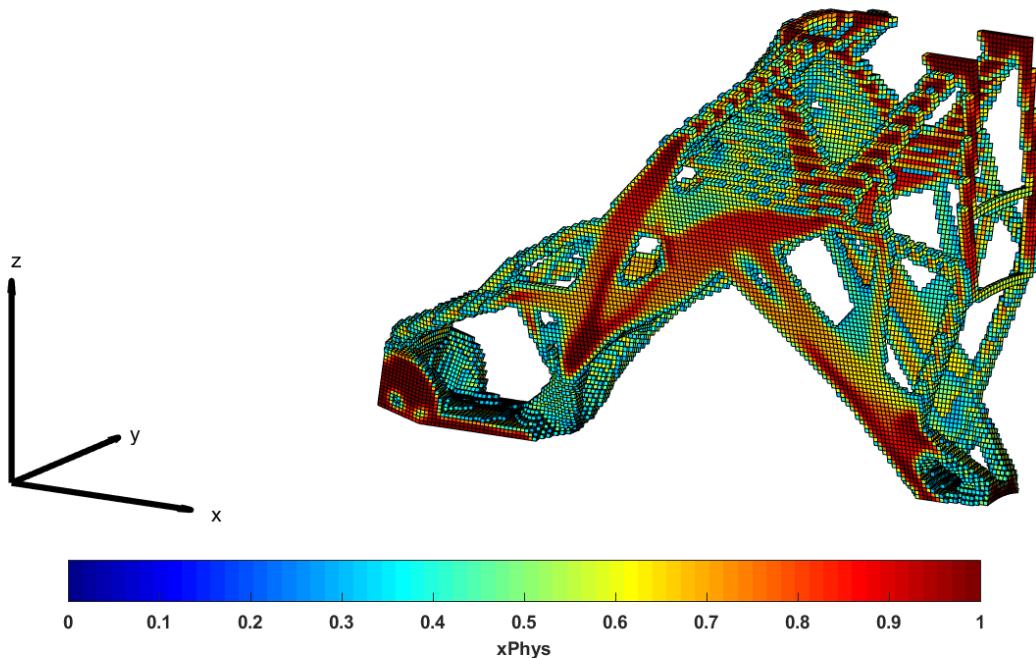


Figure B.8: Density plot of solution C obtained with  $T_0 = 0.18$

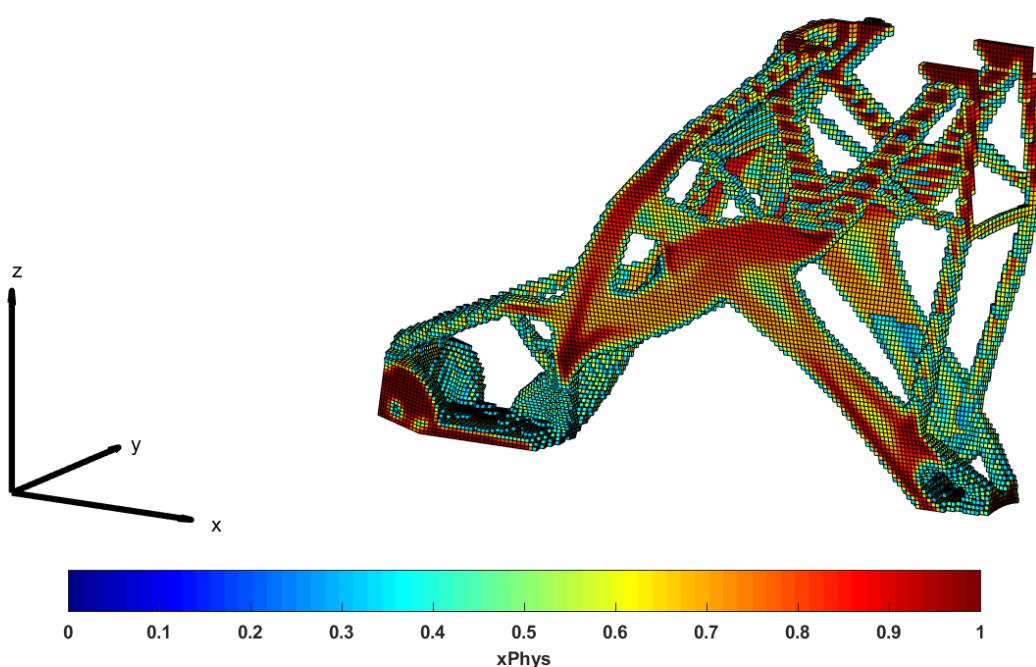


Figure B.9: Density plot of solution D obtained with  $T_0 = 1$



## APPENDIX C

# Appendix 3

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### C.1 Characteristic function and local volume fraction sensitivity distribution

This subsection presents the distribution of both characteristic function and local volume fraction sensitivity to the design variables in the example introduced in subsection 3.7. The effect of both the sampling window size and the number of Gauss point is analyzed to compute  $\delta$  from the same  $W$ . An important observation is that by increasing  $N_{GP}$  one increases the ability of GGP to adequately capture the narrow distribution of characteristic function sensitivity.

### C.2 Parametric study results on the cantilever beam case

In this section the full plot results from the parametric study on the cantilever beam presented in subsection 3.8.1 are provided.

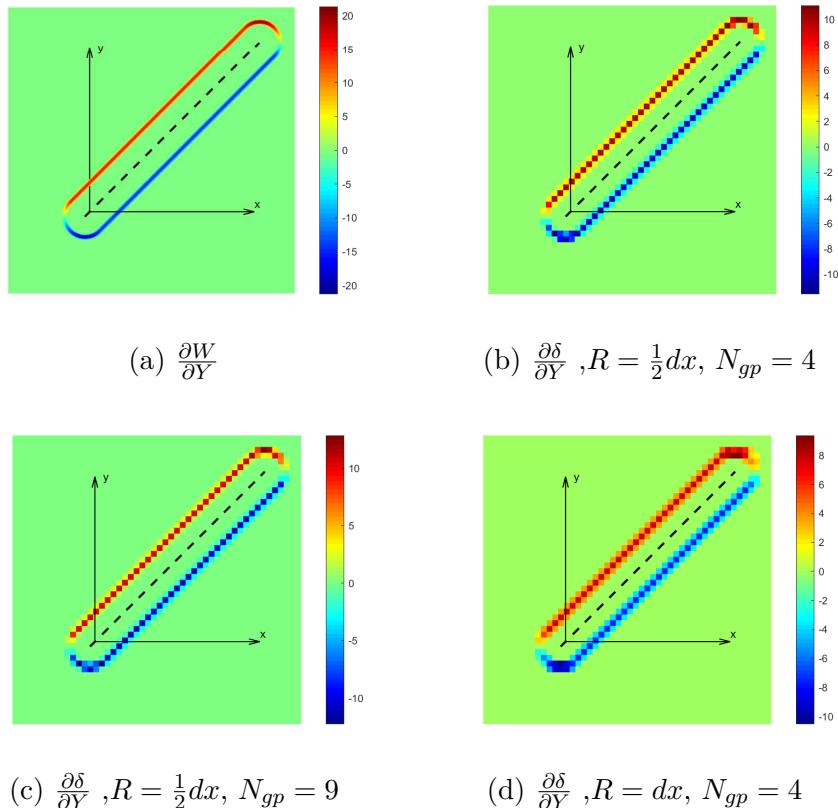


Figure C.1: Sensitivity distribution of  $W$  and  $\delta$  with respect to  $Y$  for varying number of Gauss Points  $N_{GP}$  in the sampling window and varying sampling window size  $R$ . Adapted Moving Node Approach was considered with the generic component of figure 3.1a in the configuration  $X = 1, Y = 1, L = 3, h = 0.5, \theta = \frac{\pi}{4}, \gamma = 3, \varepsilon = 0.07$  and  $dx = 0.07$  for a  $50 \times 50$  mesh over the domain of  $X_g$ .

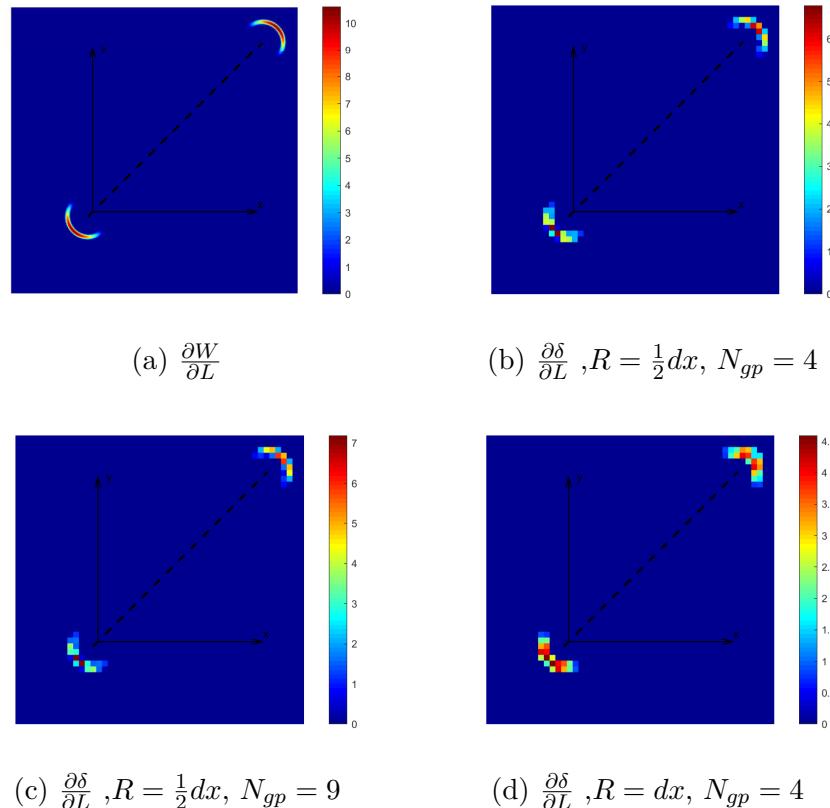


Figure C.2: Sensitivity distribution of  $W$  and  $\delta$  with respect to  $L$  for varying number of Gauss Points  $N_{GP}$  in the sampling window and varying sampling window size  $R$ . Adapted Moving Node Approach was considered with the generic component of figure 3.1a in the configuration  $X = 1, Y = 1, L = 3, h = 0.5, \theta = \frac{\pi}{4}, \gamma = 3, \varepsilon = 0.07$  and  $dx = 0.07$  for a  $50 \times 50$  mesh over the domain of  $X_g$ .

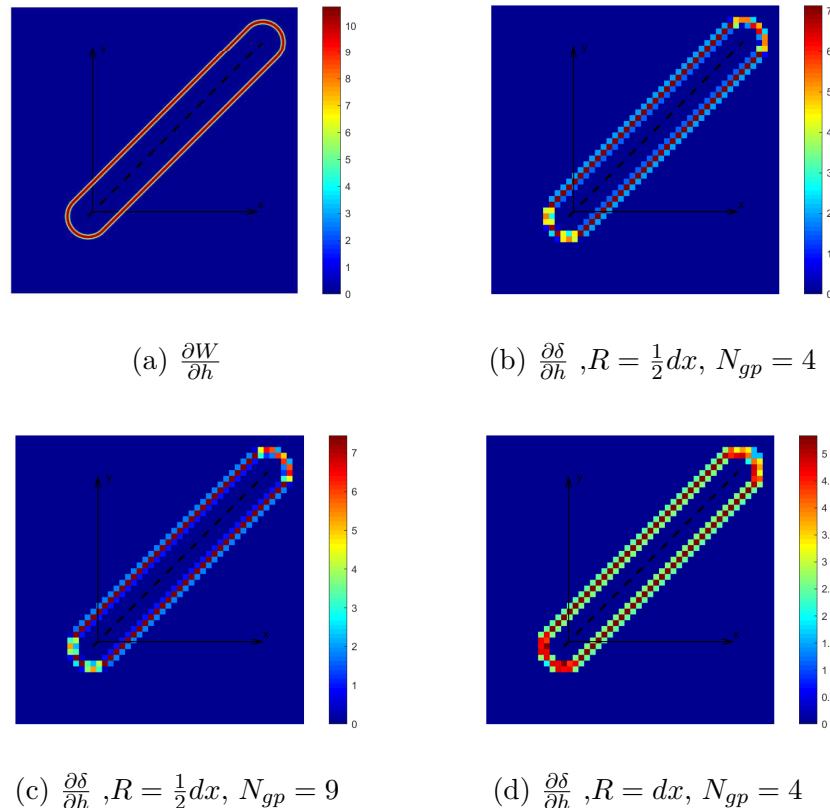


Figure C.3: Sensitivity distribution of  $W$  and  $\delta$  with respect to  $h$  for varying number of Gauss Points  $N_{GP}$  in the sampling window and varying sampling window size  $R$ . Adapted Moving Node Approach was considered with the generic component of figure 3.1a in the configuration  $X = 1, Y = 1, L = 3, h = 0.5, \theta = \frac{\pi}{4}, \gamma = 3, \varepsilon = 0.07$  and  $dx = 0.07$  for a  $50 \times 50$  mesh over the domain of  $X_g$ .

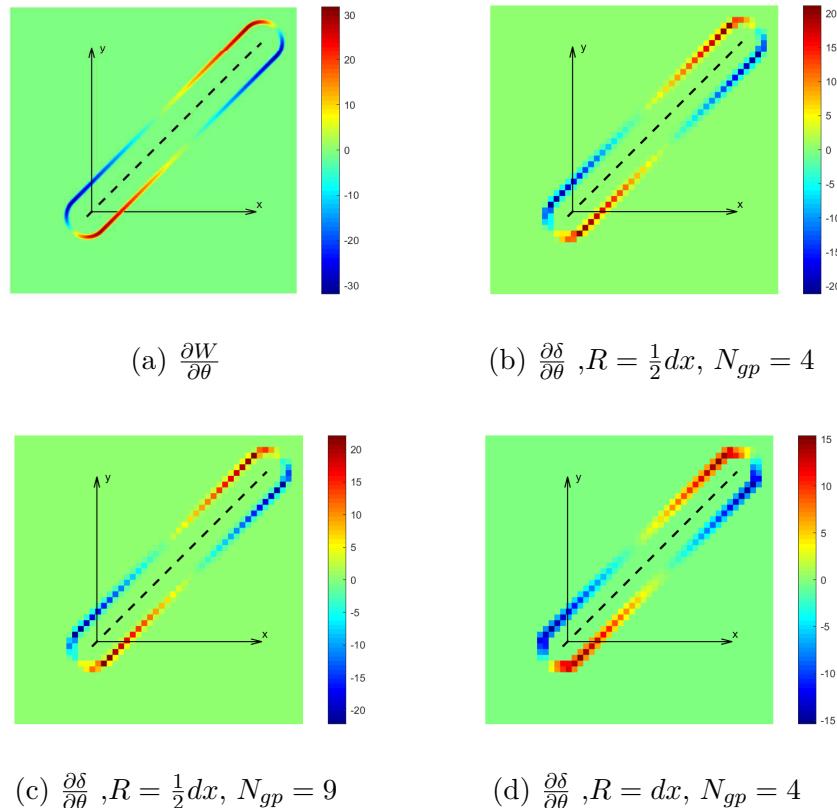


Figure C.4: Sensitivity distribution of  $W$  and  $\delta$  with respect to  $\theta$  for varying number of Gauss Points  $N_{GP}$  in the sampling window and varying sampling window size  $R$ . We considered Adapted Moving Node Approach with the generic component of figure 3.1a in the configuration  $X = 1, Y = 1, L = 3, h = 0.5, \theta = \frac{\pi}{4}, \gamma = 3, \varepsilon = 0.07$  and  $dx = 0.07$  for a  $50 \times 50$  mesh over the domain of  $X_g$ .

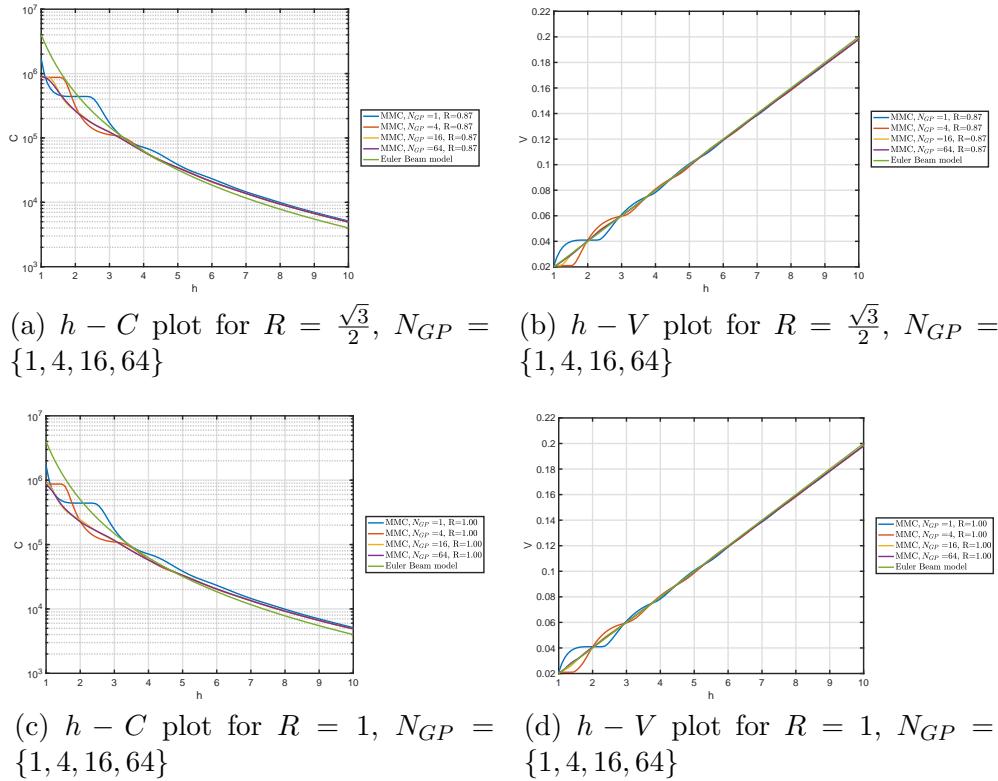


Figure C.5: Cantilever beam parametric study using the AMMC approach. Effect of the sampling window size  $R$  and of the number of Gauss points  $N_{GP}$  on the structural compliance and the volume fraction. In each graph are reported in green the true theoretical values based on the analytic beam model.

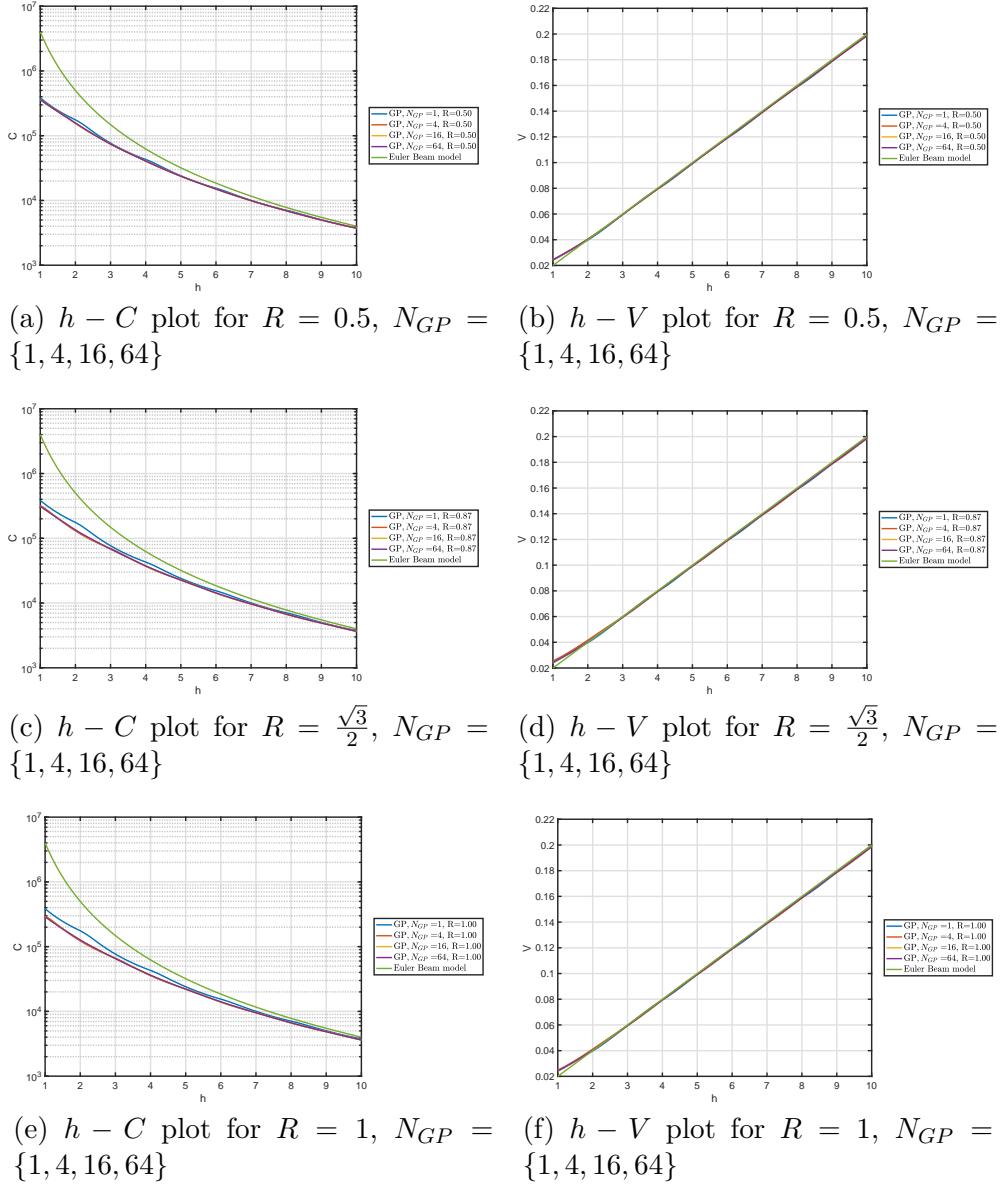


Figure C.6: Cantilever beam parametric study using AGP method. Effect of the sampling window size  $R$  and of the number of Gauss points  $N_{GP}$  on the structural compliance and the volume fraction. In each graph are reported in green the true theoretical values based on the analytic beam model.

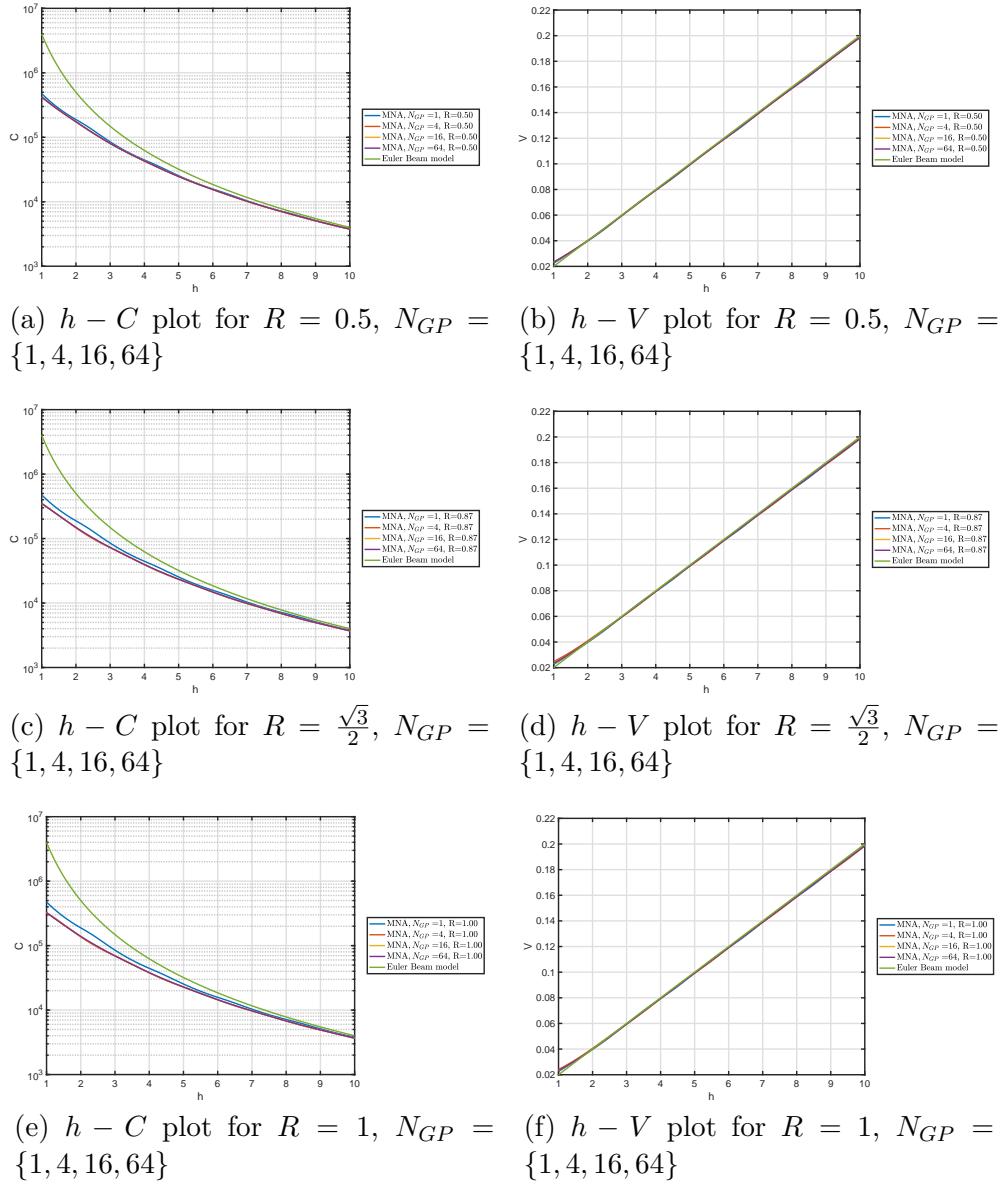


Figure C.7: Cantilever beam parametric study using the AMNA method. Effect of the sampling window size  $R$  and of the number of Gauss points  $N_{GP}$  on the structural compliance and the volume fraction. In each graph are reported in green the true theoretical values based on the analytic beam model.

### C.3 MMA set-up

This subsection provides details of implementations considered for the method of moving asymptotes (MMA, [Svanberg, 1987]). This approach makes local convex approximations at each iteration of both constraints and objective function. The convexity is adjusted by changing asymptotes' positions during the optimization history. A move limit can also be chosen in order to control the optimization step and avoid divergence. A correct scaling of both design variables and compliance is recommended to avoid numerical issues. Here it is proposed to re-scale variables and gradients according to:

$$\hat{x}_j = \frac{x_j - l_j}{u_j - l_j} \quad (\text{C.1})$$

$$\frac{dC}{d\hat{x}_j} = \frac{1}{u_j - l_j} \frac{dC}{dx_j} \quad (\text{C.2})$$

$$\frac{dV}{d\hat{x}_j} = \frac{1}{u_j - l_j} \frac{dV}{dx_j} \quad (\text{C.3})$$

where  $l_j$  and  $u_j$  are the  $j^{th}$  - component respectively of the lower bound  $\{l\}$  and of upper bound vector  $\{u\}$ . In order to avoid further MMA numerical issues one can either normalize the compliance dividing  $C$  and  $\left\{\frac{dC}{d\hat{x}}\right\}$  by a constant  $C_0$  greater than 1 that ensures the compliance and its gradient are small enough. However this way of normalizing introduces the issue of a good choice of  $C_0$ , depending on the particular problem studied. To avoid this problem, here the following normalization is considered:

$$\hat{C} = \log(1 + C) \quad (\text{C.4})$$

$$\frac{d\hat{C}}{d\hat{x}_j} = \frac{1}{1 + C} \frac{dC}{d\hat{x}_j} \quad (\text{C.5})$$

$$(\text{C.6})$$

Note that since  $C > 0$ ,  $\log(1 + C)$  is also greater than 0. This ensures the gradients to be smaller for higher values of  $C$  (that is the case of ill connected configurations). In order to avoid MMA divergence due to uncontrolled optimization step length, a strategy that is similar to the one taken by the globally convergent version of MMA (GCMMA) [Svanberg, 2002] is here provided. In the mmasub.m Matlab function called during the optimization loop modified the updating of lowmin, lowmax, uppmin and uppmax formula are modified, reducing the value of the coefficients that multiplies each variable range. This also means reducing the value set by default at 10 in equations (2.28) and (2.30). Accordingly this ensures the control of the optimization step through the overestimation of the problem convexity. In this way MMA behaves more conservatively at each iteration and is less prone to oscillate or to skip local optima<sup>1</sup>.

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<sup>1</sup>Note that this property can be beneficial or detrimental, depending on the case. Using classic MMA one can either skip worse local optima or better ones in the convergence history.



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**Résumé** — Dans cette thèse, un cadre d'optimisation topologique est développé pour améliorer la conception du mât réacteur, des supports du moteur et des nacelles. La conception optimale est obtenue en tenant en compte d'une contrainte de stress de von Mises et d'une exigence propre à la conception du moteur (c'est-à-dire une réduction de la variation des jeux en but d'aube du moteur en présence de charges de manœuvre de l'avion). Ce travail est divisé en trois parties principales. Dans la première partie, les techniques éléments finis sont passées en revue et mises en œuvre pour traiter le formalisme des superéléments et les modèles de grands espaces de conception. De plus, de nouvelles méthodes sont proposées pour traiter la liaison entre maillages à interfaces incohérentes. Dans la seconde partie, l'optimisation topologique basée sur l'approche SIMP dans un cadre eulérien est envisagée. Une stratégie multigrille est développée pour réduire le nombre d'opérations associé à la construction de la matrice de filtrage. Les défis associés aux formulations basées sur les stress sont également étudiés. Dans la troisième partie, les approches lagrangiennes d'optimisation topologique sont analysées pour les formulations à base de compliance ou à base de stress. Une nouvelle approche appelée "Generalized Geometry Projection" est proposée en tant que méthode unifiée pour la mise en œuvre de plusieurs approches lagrangiennes telles que le "Moving Morphable Components", le "Geometry Projection" et le "Moving Node Approach". La méthodologie de conception proposée a été validée sur plusieurs exemples 2D académiques, puis testée sur un modèle de moteur 3D générique (1,4 million de DDL et jusqu'à 500000 variables de conception). Le cadre proposé fournit un nouvel outil pour l'exploration de conceptions innovantes visant à améliorer l'intégration du moteur à l'aile.

**Mots clés :** Optimisation topologique, Éléments finis, Collage de maillage, Multi-maillage géométrique, Geometry Projection, Moving Morphable Components, Moving Nodes Approaches.

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**Abstract** — In this PhD thesis, a topology optimization framework is developed to support the design of pylon, engine mounts and nacelle. The optimal design is achieved considering a von Mises stress constraint and a requirement specific to engine design (i.e. reducing tip-clearance variation under aircraft maneuvers loads). This work is divided in three main parts. In the first part finite element techniques are reviewed and implemented to deal with superelement formalism and large design space models. Moreover novel methods are proposed to deal with tying of inconsistent mesh interfaces. In the second part topology optimization based on the SIMP approach is considered. A multigrid strategy is developed to reduce the computational burden associated with filter matrix construction. Challenges associated with stress based formulations are also investigated. In the third part Lagrangian approaches to topology optimization are analyzed for both compliance and stress based formulations. A novel approach called Generalized Geometry Projection is proposed as a unified method for the implementation of several Lagrangian approaches such as Moving Morphable Components, Geometry Projection and Moving Node Approach. The proposed design methodology was validated on several academic 2D examples and then tested on a generic 3D engine model (1.4 Millions of DOFs and up to 500 thousands of design variables). The proposed framework supplies a novel tool for the exploration of disruptive designs for the improvement of the engine-to-wing integration.

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**Keywords:** Topology Optimization, Finite Element Analysis, Mesh tying, Geometric Multigrid, Geometry Projection, Moving Morphable Components, Moving Nodes Approaches.

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