

Airfoil	Reynolds #	Max Cl/Cd	Description
e214-il	50,000	37.923 at $\alpha=9^\circ$	Mach=0 Ncrit=9
e214-il	100,000	61.8961 at $\alpha=7.5^\circ$	Mach=0 Ncrit=9
e214-il	200,000	87.5712 at $\alpha=5.75^\circ$	Mach=0 Ncrit=9
e214-il	500,000	121.577 at $\alpha=4.5^\circ$	Mach=0 Ncrit=9
e214-il	1,000,000	145.911 at $\alpha=4^\circ$	Mach=0 Ncrit=9