

Airfoil Tools

Search 1612 airfoils

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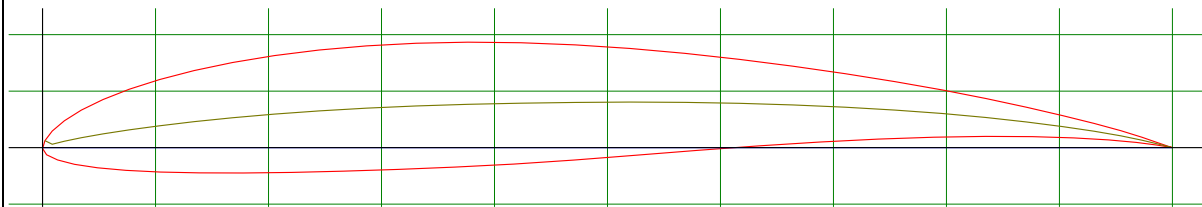
Your Reynold number range is 15,000 to 30,000. (set)

Search

E214 (11.1%) (e214-il)

E214 (11.1%) - Eppler E214 low Reynolds number airfoil





Details

(e214-il) E214 (11.1%)  
Eppler E214 low Reynolds number airfoil  
Max thickness 11.1% at 33.1% chord.  
Max camber 4% at 52% chord  
Source [UIUC Airfoil Coordinates Database](#)  
[Source dat file](#)  
The dat file is in Selig format

Dat file

E214 (11.1%)  
1.00000 0.00000  
0.99669 0.00104  
0.98737 0.00422  
0.97312 0.00916  
0.95431 0.01501  
0.93081 0.02139  
0.90279 0.02833

Parser

No parser warnings

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S4022

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[Details](#)

EPPLER 393 AIRFOIL

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GOE 547 AIRFOIL

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[Details](#)

S4158

[Preview](#)

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GOE 723 AIRFOIL

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CLARK V AIRFOIL


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NREL's S802 Airfoil

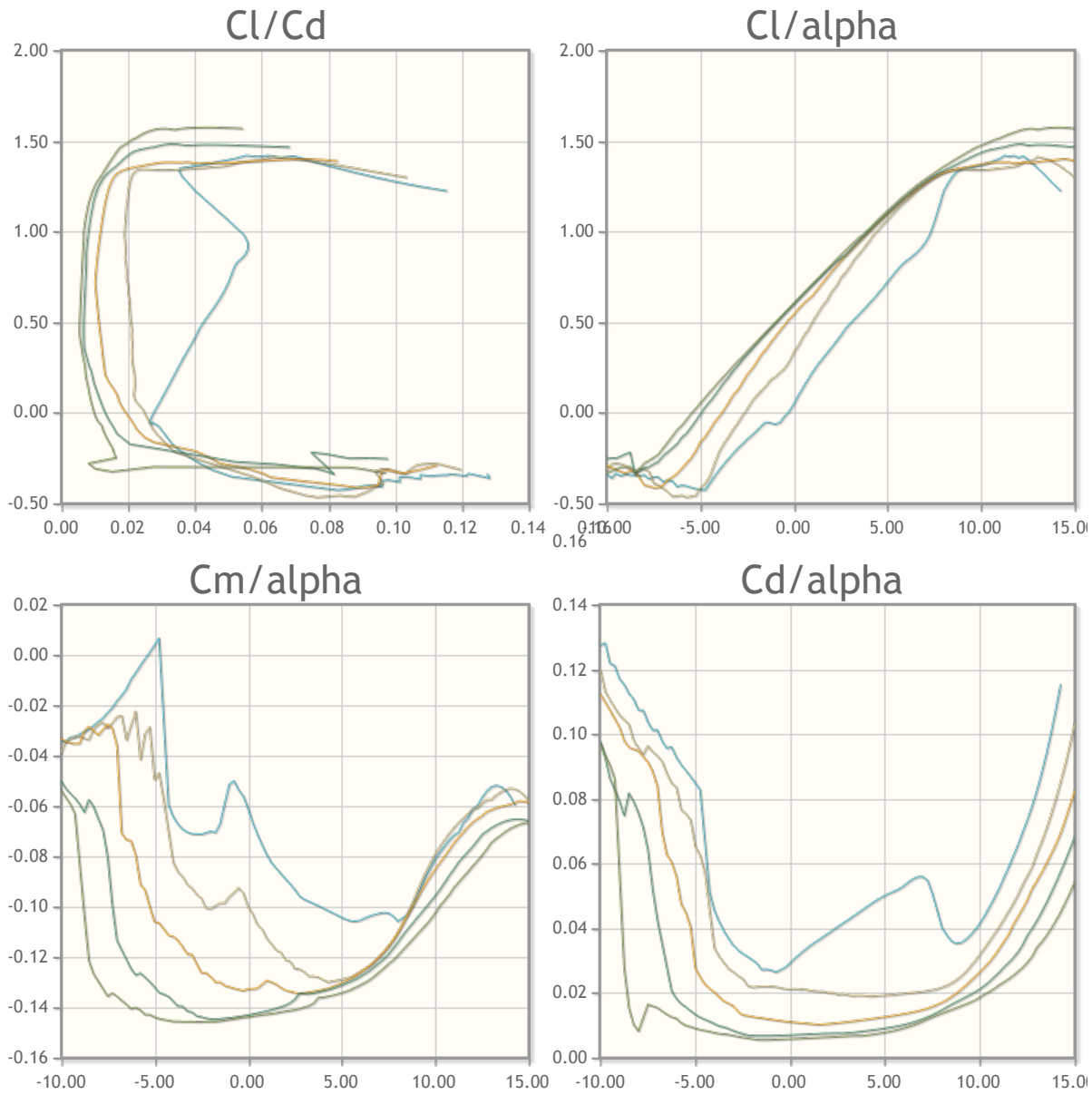
[Preview](#)

[Details](#)



Polars for E214 (11.1%) (e214-il)

Plot	Airfoil	Reynolds #	Max Cl/Cd	Description	Source
<input checked="" type="checkbox"/>	e214-il	50,000	37.923 at $\alpha=9^\circ$	Mach=0 Ncrit=9	<a href="#">Xfoil prediction</a> <a href="#">Details</a>
<input checked="" type="checkbox"/>	e214-il	100,000	61.8961 at $\alpha=7.5^\circ$	Mach=0 Ncrit=9	<a href="#">Xfoil prediction</a> <a href="#">Details</a>
<input checked="" type="checkbox"/>	e214-il	200,000	87.5712 at $\alpha=5.75^\circ$	Mach=0 Ncrit=9	<a href="#">Xfoil prediction</a> <a href="#">Details</a>
<input checked="" type="checkbox"/>	e214-il	500,000	121.577 at $\alpha=4.5^\circ$	Mach=0 Ncrit=9	<a href="#">Xfoil prediction</a> <a href="#">Details</a>
<input checked="" type="checkbox"/>	e214-il	1,000,000	145.911 at $\alpha=4^\circ$	Mach=0 Ncrit=9	<a href="#">Xfoil prediction</a> <a href="#">Details</a>
<div>Update plots</div>	Tired of clicking boxes? Set you Re# range <a href="#">here</a>		<a href="#">Reynolds number calculator</a>		

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