

# Steady-State Low-Order Explicit (LOE) Runge-Kutta Schemes with Improved Convergence

AIAA Aviation

Zaid Sabri

Ray Hixon

University of Toledo  
Toledo Ohio, 43606 USA

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# Background

- Computational Fluid Dynamics (CFD) is the study of fluid mechanics that predicts fluid flows using digital computers.
- To accomplish this goal, the analytic governing equations of fluid motion are discretized in space (using structured grid meshing), resulting in a system of coupled ordinary differential equations (ODE).



# Implicit Time Marching

- The one-dimensional Euler equation can be written as:

$$\frac{\partial Q}{\partial t} + \frac{\partial E}{\partial x} \Big|^{n+1} = 0 \quad (1)$$

- In these equations, the flux  $E$  is a nonlinear function of the flow vector  $Q$ .
- We linearize the Taylor Series expansion of the fluxes, to rewrite the delta form of the equation as:

$$\underbrace{\left[ I + \frac{\partial}{\partial x} \left( \Delta t \cdot \frac{\partial E}{\partial Q} \Big|_i^n \right) \right]}_{\text{Numerics}} \{ \Delta Q_i \} = - \Delta t \cdot \underbrace{\left\{ \frac{\partial}{\partial x} (E_i^n) \right\}}_{\text{Physics}} \quad (2)$$

# Implicit Time Marching - Preconditioning

- Instead of using:

$$\left[ I + \Delta t \left( \frac{A_{i+3} - 8A_{i+2} + 37A_{i+1} - 37A_{i-1} + 8A_{i-2} - A_{i-3}}{48\Delta x} \right)^{n+1,l} \right] \Delta Q_i = -\Delta t \{RHS\}_i^{n+1,l} \quad (3)$$

is preconditioned as:

$$\left[ I + \Delta t \left( \frac{A_{i+1} - A_{i-1}}{2\Delta x} \right)^{n+1,l} \right] \Delta Q_i = -\Delta t \{RHS\}_i^{n+1,l} \quad (4)$$

- This will result in a more efficient solver since the number of diagonals have been reduced.

# Stability

- The equation is changed to add a scaling factor  $\sigma$  multiplying the spatial derivative on the LHS:

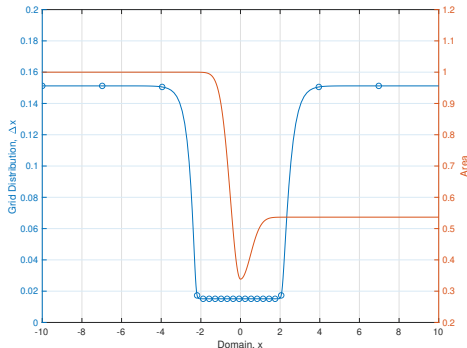
$$\left[ 1 + \Delta t \sigma \left( \frac{A_{i+1} - A_{i-1}}{2\Delta x} \right)^{n+1,l} \right] \Delta Q_i = - \Delta t \{RHS\}_i^{n+1,l} \quad (5)$$

	$\sigma_{E2}$	$\sigma_{E4}$	$\sigma_{E6}$	$\sigma_{DRP}$	$\sigma_{RDRP}$	$\sigma_{C4}$
<b>Periodic</b>	1.0	1.0	11/10	1.16682	7/6	1.49982
<b>Bounded</b>	1.0	2.121796	6.352796	4.9748714	6.3620687	3.80018

# Numerical Results for Benchmark Problems

- One Problem from the Third Computational Aeroacoustics (CAA) workshop is analyzed to validate the work done
- The Quasi-1D nonlinear Euler equations are used to solve this problem. The equations are given in the conservative variables as:

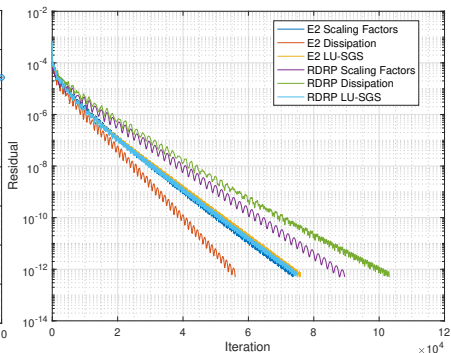
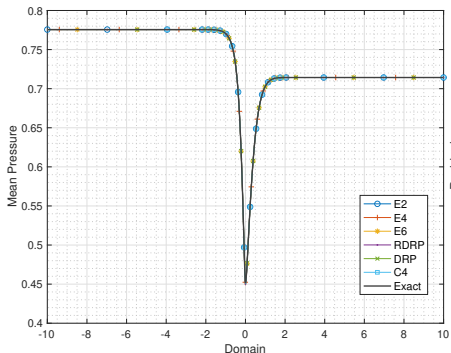
$$\left( \begin{array}{l} \frac{\partial}{\partial t} \left\{ \begin{array}{l} \rho \\ \rho u \\ E_{tot} \end{array} \right\} \\ + \frac{\partial}{\partial x} \left\{ \begin{array}{l} \rho u \\ \rho u^2 + p \\ u(E_{tot} + p) \end{array} \right\} \\ + \frac{1}{A} \frac{\partial A}{\partial x} \left\{ \begin{array}{l} \rho u \\ \rho u^2 \\ u(E_{tot} + p) \end{array} \right\} \end{array} \right) = 0$$



# Category 1 Problem 1, Steady State Results

- The mean flow is set as:

$$\begin{Bmatrix} \bar{\rho} \\ \bar{u} \\ \bar{p} \end{Bmatrix} = \begin{Bmatrix} 1.0 \\ 0.4 \\ \frac{1}{\gamma} \end{Bmatrix}$$

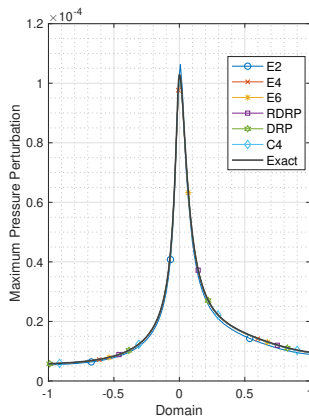
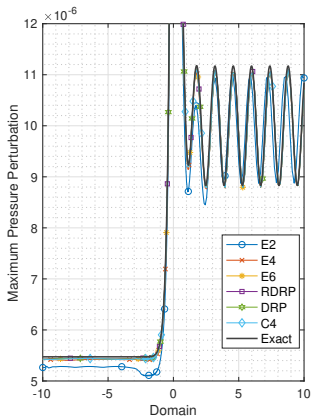




# Category 1 Problem 1, Unsteady Results

Table: CFL Values for Category 1 Problem 1

	<b>E2</b>	<b>E4</b>	<b>E6</b>	<b>DRP</b>	<b>RDRP</b>	<b>C4</b>
$\nu_{physical}$	12.521	17.518	19.858	20.834	20.834	22.237



# Conclusion

- The present research aimed to establish a highly efficient implicit solver while using higher-order spatial differencing schemes.
- Through scaling factors, the LHS matrix was replaced with an easier to solve matrix, cutting down on the computational work needed for the iterative steady-state solver.
- Although the current study is based on one-dimensional problems, the findings suggest stability is possible if scaling factors are used on the LHS spatial difference.
- The preconditioned matrix was tested against steady and unsteady CAA workshop data; the numerical solution agrees well with the exact.



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Zaid Sabri

University of Toledo

Toledo Ohio, 43606 USA

+1 (567)377-9009

zaid.sabri@rockets.utoledo.edu

