# Designing an MPC Controller for a Simplified Rocket Landing Problem

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Index Terms—MPC, Rocket Landing, Model Predictive Control, Control Theory, Control Engineering

#### I. INTRODUCTION

The objective of the assignment is to implement a model-predictive control (MPC) scheme to land a simplified model of a rocket. The problem is fundamentally one of 'controllability', ie  $x(0) = x_s \rightarrow x(t_f) = 0$ . The assignment gives an abstracted model, we start by rewriting the model as follows:

$$\begin{bmatrix} r_x(k+1) \\ r_y(k+1) \\ r_z(k+1) \\ v_x(k+1) \\ v_y(k+1) \\ v_z(k+1) \end{bmatrix} = \begin{bmatrix} 1 & 0 & 0 & T & 0 & 0 \\ 0 & 1 & 0 & 0 & T & 0 \\ 0 & 0 & 1 & 0 & 0 & T \\ 0 & 0 & 0 & 1 & 0 & 0 \\ 0 & 0 & 0 & 0 & 1 & 0 \\ 0 & 0 & 0 & 0 & 0 & 1 \end{bmatrix} \begin{bmatrix} r_x(k) \\ r_y(k) \\ r_z(k) \\ v_x(k) \\ v_y(k) \\ v_y(k) \\ v_z(k) \end{bmatrix} +$$

$$\begin{pmatrix} \frac{T^2}{2m} & 0 & 0\\ 0 & \frac{T^2}{2m} & 0\\ 0 & 0 & \frac{T^2}{2m}\\ T & 0 & 0\\ 0 & T & 0\\ 0 & 0 & T \end{pmatrix} \begin{bmatrix} f_x(k) + w_x\\ f_y(k) + w_y\\ f_z(k) - mg \end{bmatrix}$$

Although we assume the system is reachable, it is important to check for controllability. Through MATLAB: Rank = rank(ctrb(A, B)) we find the system is controllable, and we can proceed with designing a controller.

#### II. DESIGN

#### A. Controller Design

Three separate controllers are considered in this report. The first controller is an unconstrained, two stage MPC. The second controller builds on the first, and introduces both state and input constraints. The third controller further builds on

the first two, introducing disturbance rejection. We start by looking at the unconstrained controller.

1) Unconstrained: The Q and R matrices are chosen as follows:

$$Q = \begin{bmatrix} 5 & 0 & 0 & 0 & 0 & 0 \\ 0 & 5 & 0 & 0 & 0 & 0 \\ 0 & 0 & 100 & 0 & 0 & 0 \\ 0 & 0 & 0 & 1 & 0 & 0 \\ 0 & 0 & 0 & 0 & 1 & 0 \\ 0 & 0 & 0 & 0 & 0 & 100 \end{bmatrix}; R = I \cdot 0.1 \text{ for } I \in \mathbb{R}^{3 \times 3}$$

The Q matrix was chosen as to prioritose the vertical position and velocity. This represents the high level of importance given to landing at the correct altitude. Similarly, the prioritisation of velocity represents the importance of landing a velocity that prevents a hard landing. The lateral positions, ie  $r_x$  and  $r_y$ , are given a lower priority than the vertical position, but a higher priority than the lateral velocity,  $v_x$  and  $v_y$ . The reasoning behind this is that it is more important that the landing location is correct over the lateral velocities throughout the entire simulation.

The R matrix was chosen as to penalise the control inputs,  $f_x$ ,  $f_y$  and  $f_z$ , equally. The small value given to R, 0.1, represents the low cost associated with expending fuel to control the rocket, and encourages the rocket to use fuel to exercise control in all three directions equally. Essentially, control is cheap. The horizon length N was chosen to be 5 as it provided a spectral radius of 0.5975, which is well within the stability region of the system.

2) Constrained: The constrained controller builds on the unconstrained controller by introducing both state and input constraints. The input constraints are chosen as follows:

$$u_{max} \begin{bmatrix} \frac{f_z}{\tan \theta} + 50\\ \frac{f_z}{\tan \theta} + 50\\ 0 \end{bmatrix}; u_{min} = \begin{bmatrix} -\frac{f_z}{\tan \theta} + 50\\ -\frac{f_z}{\tan \theta} + 50\\ -12 \end{bmatrix}$$

where  $\theta=30^\circ=0.52$  rad. Theta is chosen as a constant as this is the maximum allowable angle of the rocket engines.

Figure ??tik:axes shows the justification for setting the  $f_z$  constraint as negative, rather than positive.



Fig. 1.  $f_z$  on the z axis.

The state constraints are chosen as follows:

$$x_{max} = \begin{bmatrix} 600\\600\\500\\5\\5\\0 \end{bmatrix}; x_{min} = \begin{bmatrix} -600\\-600\\0\\-5\\-5\\-15 \end{bmatrix}$$

### 3) Disturbance Rejection:

#### B. Experiment Setup

First, it is important to state simplifications assumed. Let the starting parameters be as follows:

$$\begin{vmatrix} r_x(0) \\ r_y(0) \\ r_z(0) \\ v_x(0) \\ v_y(0) \\ v_z(0) \end{vmatrix} = \begin{vmatrix} 600 \\ 600 \\ 500 \\ 5 \\ 5 \\ -15 \end{vmatrix} \land f_x(0) = f_y(0) = f_z(0) = 0$$

These starting conditions are the limits of what the assignment permits, i.e a maximum starting altitude of 500m, and a maximum lateral distance of 600m.

# III. RESULTS IV. ANALYSIS & DISCUSSION V. CONCLUSION

#### VI. EASE OF USE

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$$a + b = \gamma \tag{2}$$

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# Thrust Control Inputs $[u_x, u_y, u_z]$

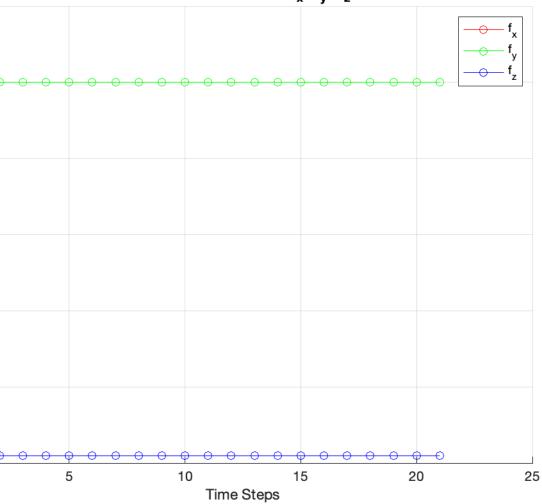


Fig. 2. Example of a figure caption.

figures and tables after they are cited in the text. Use the abbreviation "Fig. ??", even at the beginning of a sentence.

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