List of Figures

List of Tables

Nomenclature

A = nozzle cross-sectional area

H = nozzle height

M = Mach number

NPR = nozzle pressure ratio, P_0/P_a

P = pressure

P_0 = total pressure at the nozzle inlet

T = temperature

u,v,w = velocity components

x = axial direction

y = normal direction

γ = ratio of specific heats

θ = flow angle

μ = viscosity

φ = shock angle

a = ambient

c = centerline

e = nozzle exit

t = throat

Contents

Abstract

ABSTRACT OF YOUR SEMINAR WORK GOES HERE

1 Introduction

INTRODUCTION OF YOUR SEMINAR WORK GOES HERE

1.1 Scope & Methodology

YOUR SEMINAR WORK GOES HERE

2 Literature Review

The review should be conducted from at least five research papers published during last five year.

YOUR SEMINAR WORK GOES HERE

3 Case study

YOUR SEMINAR WORK GOES HERE

4 A Sample Section

4.1 Title of Sample Subsection

For a figure sample with caption and proper reference, see Figure 1 as adopted from[1]. The figure number and reference numbers are automatically generated in a chronological order by LaTeX.

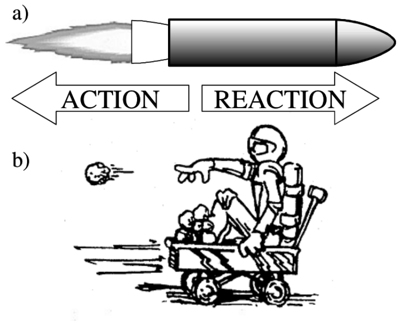

Figure 1: Here Goes Your Figure's Caption.

A sample equation is written as:

$$F_m = \frac{dm}{dt}v_e = \dot{m}v_e \tag{1}$$

where m is the mass flow rate and v e is the exit or exhaust velocity of the propellant.

An another sample equation can be expressed as

$$F_m = \dot{m}v_e + (P_e - P_a)A_e \tag{2}$$

where p_e and A_e are the pressure and cross section area at the nozzle exit, and p_a is the ambient pressure.

5 Conclusion

CONCLUSION, IF ANY.

6 Future Work

FUTURE WORK, IF ANY

References

Humble R W, Henry G N and Larson W J (1995), *Space propulsion analysis and design*, McGraw-Hill, Inc., ISBN-0-07- 031329-6.

Author First, Author Second *Title of the paper* Name of Journal Page-numbers, Month Year.