# Model Documentation of the 'CCV-type aircraft'

#### 1 Nomenclature

#### 1.1 Nomenclature for Model Equations

- x state vector
- u control input vector
- w noise vector
- z regulated output vector
- y measurement vector

### 2 Model Equations

State Vector and Input Vector:

$$x \in \mathbb{R}^5 u$$
  $\in \mathbb{R}^2 w \in \mathbb{R}^5 z$   $\in \mathbb{R}^5 y \in \mathbb{R}^4$ 

System Equations:

$$\dot{x}(t) = Ax(t) + B_1 w(t) + Bu(t) \tag{1a}$$

$$z(t) = C_1 x(t) + D_{11} w(t) + D_{12} u(t)$$
(1b)

$$y(t) = Cx(t) + D21w(t)$$
(1c)

Outputs: z

#### 2.1 Exemplary parameter values

Symbol	Value
A	$\begin{bmatrix} -1.341 & 0.9933 & 0 & -0.1689 & -0.2518 \end{bmatrix}$
	$\begin{vmatrix} 43.223 & -0.8693 & 0 & -17.251 & -1.5766 \end{vmatrix}$
	1.341 0.0067 0 0.1689 0.2518
	0   0   0   -20.0   0
	$\begin{bmatrix} 0 & 0 & 0 & -20.0 \end{bmatrix}$
В	
	20.0 0
$B_1$	
	20.0 0
$C_1$	$\begin{bmatrix} 1.0 & 0 & 0 & 0 & 0 \end{bmatrix}$
	0 1.0 0 0 0
	0 0 1.0 0 0
	0 0 0 1.0 0
	$\begin{bmatrix} 0 & 0 & 0 & 0 & 1.0 \end{bmatrix}$
C	$\begin{bmatrix} 0 & 0 & 1.0 & 0 & 0 \end{bmatrix}$
	$\begin{vmatrix} 47.76 & -0.268 & 0 & -4.56 & 4.45 \end{vmatrix}$
	0 0 0 1.0 0
	$\begin{bmatrix} 0 & 0 & 0 & 0 & 1.0 \end{bmatrix}$
$D_{11}$	$\begin{bmatrix} 0 & 0 & 0 & 0 & 0 \end{bmatrix}$
	$[0 \ 0 \ 0 \ 0 \ 0]$
$D_{12}$	
	1.0 0
	0 1.0
$D_{21}$	
	<u> </u>

# 3 Derivation and Explanation

This model is part of the "'COMPleib"' - library and was automatically imported into ACKREP.

The original description was:

AC11 CCV-type aircraft A. T. Alexandridis and P. N. Paraskevopoulos, "A New Approach to Eigenstructure Assignment by Output Feedback", TOAC, Vol. 41, Nr. 7, pp. 1046-1050, 1996 Example 2

## 4 Simulation

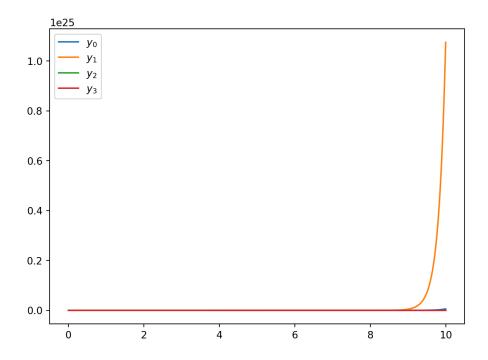


Figure 1: Simulation of the CCV-type aircraft.

## References

[1] . T. Alexandridis and P. N. Paraskevopoulos, "A New Approach to Eigenstructure Assignment by Output Feedback", TOAC, Vol. 41, Nr. 7, pp. 1046-1050, 1996 Example 2