

Energy-Aware Dynamic Planning Algorithm for Autonomous UAVs

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I. INTRODUCTION

Many scenarios involving unmanned aerial vehicles (UAVs), such as precision agriculture, search and rescue, and surveillance, require high autonomy but have limited energy budgets. A typical example of these scenarios is a UAV flying a path and performing some on-board computational tasks. For instance, the UAV might detect ground patterns and inform other ground-based actors with little human interaction. We refer to the computational tasks that can be dynamically replanned as computations. We are interested in energy-aware dynamic planning, which would find optimal tradeoffs between the path, computations, and energy requirements. Current generic planning solutions for outdoor UAVs do not plan path and computations dynamically, nor are energy-aware. They are often semi-autonomous; the plan is static and usually defined using planning software [1]. Such a state of practice has prompted us to propose an *energy-aware dynamic planning algorithm* for UAVs. The algorithm combines and generalizes some of the past body of knowledge on mobile robot planning problems and addresses the increasing *computational demands* and their relation to energy consumption, path, and autonomy for the UAV planning problem.

Planning algorithms literature for mobile robots includes topics such as trajectory generation and path

planning. Generally, the algorithms select an energy-optimized trajectory [2], e.g., by maximizing the operational time [3]. However, they apply to a small number of robots [4], and focus exclusively on planning the trajectory for these robots [5], despite compelling evidence for the energy consumption also being significantly influenced by computations [6]. Given the availability of powerful GPU-equipped mobile hardware [7], the use of computations is expected to increase in the near future [8]–[10]. More complex planning, which includes a broader concept of the plan being a set of tasks and a path, all focus on the trajectory [6], [11] and apply to a small number of robots [12], [13]. For UAVs specifically, rotorcrafts have gained interest in terms of algorithms for energy-optimized trajectory generation [14], [15].

Unlike most of the past planning algorithms literature, our algorithm plans both the path and computations. We guide the UAV using a vector field [16] that converges smoothly to a planned trajectory. The use of vector fields for planning is widely discussed in the literature [17]–[21]. We denote the trajectory with a mathematical function that we refer to as the trajectory explicit equation (TEE). The path is composed of multiple TEEs. We express the computations with a modeling tool presented in our previous work [22]. The algorithm alters the TEE and computations within some parameters using output model predictive control (MPC). These parameters are the maximum, minimum, and current levels of the TEE and computations. The user specifies the parameters in an initial plan with an energy budget. The energy budget is the battery capacity and some other battery parameters. In the remainder of the paper, we adopt the following notation. We refer to the current TEE and computations parameters simply as TEE and computations parameters. We refer to both as parameters. We refer to the constraints sets that delimit the maximum and minimum levels of the parameters as TEE sets and computations sets. Again, we refer to both as parameters sets. There can be multiple or none computations sets per each task, and the same applies to the TEE. Our goal is to complete the plan with the highest possible parameters as the UAV flies and its batteries drain. The term parameters should not be confused with the battery parameters, which we always mention explicitly. These are fixed values that are not replanned by the algorithm.

To achieve the energy-aware dynamic planning, we further introduce and formally proof a periodic energy

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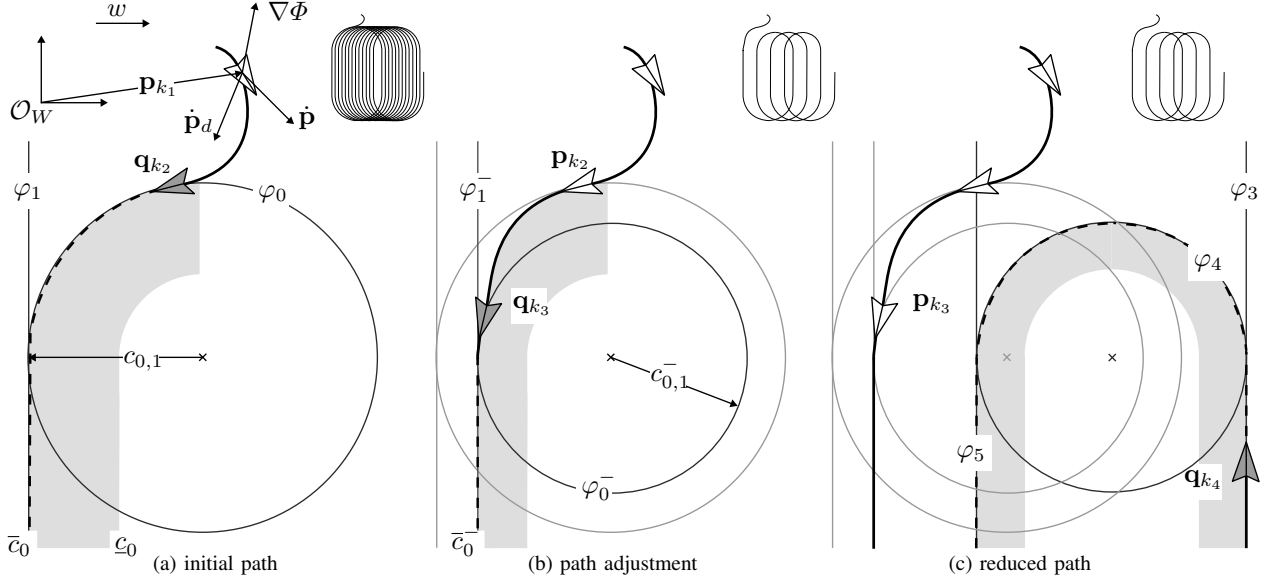


Fig. 1. The mission consists of circles and lines covering a polygon. The UAV heads to φ_0 initially, using the desired velocity vector $\dot{\mathbf{p}}_d$. It later heads to φ_0^- reducing the radius $c_{0,1}$ to satisfy the battery constraints. The UAV then converges to φ_1 in stage \mathcal{M}_1 , φ_2 in stage \mathcal{M}_2 , and so on (the circle φ_2 is not visible in the figure; it connects φ_1 and φ_3).

model that accounts for the uncertainty (atmospheric interference). We use Fourier analysis to derive the model, and state estimation to address the uncertainty. Periodicity is often present due to the path and computations repetitive patterns [23]. Indeed UAV scenarios often iterate over a set of tasks and trajectories (e.g., monitoring or search and rescue). Given that the plan is periodic, we expect the energy to approximately evolve periodically. The observation applies broadly; we show different UAV scenarios with periodic energy signals. Moreover, we briefly investigate such signals in some ground-based mobile robots.

In the spirit of reducing costs and resources, we showcase the algorithm using the problem of dynamic planning for a precision agriculture fixed-wing UAV. Precision agriculture is often put into practice [24] with ground mobile robots used for harvesting [25]–[30], and UAVs for preventing damage and ensuring better crop quality [1], [31]. The plan is structured as follows. Trajectory-wise, the UAV flies in circles and lines covering a polygon. Computationally-wise, it detects obstacles using a convolutional neural network (CNN) and informs grounded mobile robots employed for harvesting. The algorithm alters the plan; it controls the processing rate and the radius of the circles (affecting the distance between the lines). Figure 1 shows an initial slice of such a plan. The UAV first heads to a circular TEE with a given radius, which is later reduced as, e.g., windy weather requires the adjustment of the control input to avoid battery depletion. Computations significantly impact the battery, with a potential extension of

up to 13 minutes over an hour by merely switching to the lowest computations (see Section IV).

The remainder of the paper is organized as follows. The overview of dynamic planning is provided in Section II, along with a suitable model for the position and energy. The algorithm is proposed in Section III. Section IV presents the result and showcases the performances. We propose some conclusions in Section V. Finally, we provide some additional information in the Appendix I and examples in the Appendix IV.

II. PLANNING OVERVIEW

We assume that a user-specified initial plan consists of different stages. At each stage, the UAV must follow the TEE within TEE sets and do the computations within the computations sets.

Let us suppose the TEE at stage i can be altered with ρ TEE parameters and the computations with σ computations parameters. We then express the TEE as a continuous twice differentiable function $\varphi_i : \mathbb{R}^2 \times \mathbb{R}^\rho \rightarrow \mathbb{R}$ of a point and the TEE parameters which returns a metric of the distance between the point and the nominal trajectory. We express and the computations as the value of the computations parameters. We discuss the concrete meaning of the value of TEE parameters in Subsection II-B), and the of the computations parameters in Subsection II-D).

Let us further adopt the following mathematical notation. $[a]$ is the set $\{0, 1, \dots, a\}$, $[a]^+$ the set $[a]/\{0\}$, $\langle a_1, a_2, \dots, a_n \rangle$ an ordered list of n elements, c^0 the optimal value of parameter c , and \underline{c}, \bar{c} its lower and upper

bound retrieved from a plan specification. The plan specification is a file which specifies all the stages and triggering points along the plan. We discuss the format in Subsection II-A. Moreover, we use the notation $c_{i,j}$ to specify the j -th item of the i -th parameters set c_i .

Definition II.1 (Stage, plan and triggering point). A stage Γ_i at time instant k of a plan Γ is defined as the ordered list

$$\begin{aligned} \Gamma_i := \{ & \langle \varphi_i(\mathbf{p}_k, c_{i,1}, \dots, c_{i,\rho}), c_{i,\rho+1}, \dots, c_{i,\rho+\sigma} \rangle \\ & | \exists \mathbf{p}_k, \varphi_i(\mathbf{p}_k, c_{i,1}, \dots, c_{i,\rho}) \in \mathbb{C}_i, \\ & \forall j \in [\sigma]^+, c_{i,\rho+j} \in \mathbb{S}_{i,j} \}, \end{aligned} \quad (1)$$

where $\mathbb{C}_i := [\underline{c}_i, \bar{c}_i] \subseteq \mathbb{R}$ is the TEE set, and $\mathbb{S}_{i,j} := [\underline{c}_{i,\rho+j}, \bar{c}_{i,\rho+j}] \subseteq \mathbb{Z}_{\geq 0}$ the j -th computation set. \mathbf{p}_k is a point of a UAV flying at an assigned altitude $h \in \mathbb{R}_{>0}$ w.r.t. some inertial navigation frame \mathcal{O}_W . We note that while the TEE can differ for each stage, the tasks remain the same. However, the user can inhibit or enable a computation varying its computation set.

The *plan* is a finite state machine Γ where the state-transition function $\bigcup_i \Gamma_i \times \mathbb{R}^2 \rightarrow \bigcup_i \Gamma_i$ maps a stage and a point to the next stage. The point \mathbf{p}_{Γ_i} that allows the transition between Γ_i and Γ_{i+1} is called *triggering point*.

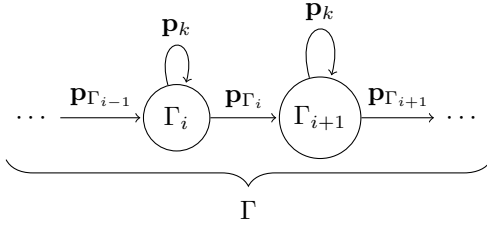


Fig. 2. Finite state machine of the plan

A slice of a finite state machine in Figure 2 shows the transition between the stages. The triggering point \mathbf{p}_{i-1} allows the transition to stage Γ_i . The UAV remains in the stage with any generic point \mathbf{p}_k . It eventually enters the stage Γ_{i+1} with the triggerint point \mathbf{p}_i and so on, until it reaches the last triggering point.

The algorithm takes as input the plan, the triggering points (using the plan specification) and some initial guesses that are necessary for the position and energy estimation. It outputs the new position, the instantaneous energy consumption, and the control input—an action performed evolving the state.

A. Plan specification file format

We specify the stages $\bigcup_i \Gamma_i$, triggering points $\bigcup_i \mathbf{p}_{\Gamma_i}$, and the plan Γ using a specific file format—the plan specification. We refer the reader to Appendix II for an example.

The first stage Γ_1 is specified as the first line of the file

$$1: k_e; r; \varphi_1(\mathbf{p}_k, c_{1,1}, \dots, c_{1,\rho})$$

where k_e is a convergence value, r a rotation, and φ_1 the TEE. We refer the reader to the next subsection where we discuss the physical meaning of the convergence value and rotation. The rotation is expressed in binary format with zero being the counter-clockwise rotation, one the clockwise rotation.

The next line specifies the triggering point \mathbf{p}_{Γ_1}

$$2: x, y$$

where x and y are the two coordinates of the point. The UAV reaches the point within a given radius (a parameter of the algorithm) and enters the next stage Γ_2 . The coordinates can be expressed in function of the TEE parameters c_1 . Generally, one can express the coordinates in function of any TEE parameters c_i , propagating the information therein if necessary (see Figure 3).

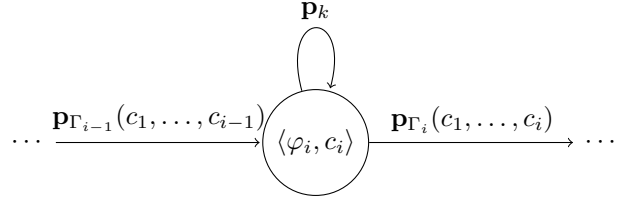


Fig. 3. Detail of the stage Γ_i in the finite state machine

We refer the reader to the example in Appendix II for a detailed implementation. In the example, the first TEE parameter is propagated to all the following TEEs and triggering points.

The following $\rho + \sigma$ lines

$$3: [\underline{c}_i, \bar{c}_i]$$

specifies the lower \underline{c}_i and upper \bar{c}_i bound of the i -th parameter set. They are in ascending order and the TEE parameters sets are specified before the computations parameters sets.

Follows the entries for stage Γ_2 (equivalent to lines 1–3), Γ_3 and so on, until the last stage. We assume that the last triggering point is the final UAV position.

B. State: position and energy

We describe the position with x, y coordinates and the altitude h , and the energy with some energy coefficients which describes the energy evolution in time. Here we describe the position evolution using vector field [16].

Let us assume the set

$$\mathcal{P}_i := \{\mathbf{p}_k \mid \varphi_i(\mathbf{p}_k, c_{i,1}, \dots, c_{i,\rho}) \in \mathbb{C}_i\}, \quad (2)$$

delimits the area where the i -th TEE φ_i is free to evolve using ρ adjustments $\mathbf{c}_i := c_{i,1}, \dots, c_{i,\rho}$ (the gray

area in Figure 1). Physically, φ_i is function of the two coordinates and is satisfied when a point \mathbf{p}_k is on the TEE. The TEE parameters allows to change the TEE. They are a way to alter the nominal trajectory from the initial plan.

In fact, the algorithm uses the set from Equation (2) to alter the optimal TEE parameters \mathbf{c}_i^0 s.t. $\varphi_i(\mathbf{p}_k, \mathbf{c}_i^0)$ has the highest instantaneous energy consumption (while still respecting the energy budget). It guides the UAV to the new position \mathbf{p}_{k+1} using the function $\Phi := \varphi_i(\mathbf{p}_k, \mathbf{c}_i^0)$, computing its vector field $\nabla\Phi := (\partial\Phi/\partial x, \partial\Phi/\partial y)$, and deriving the direction to follow in the form of velocity vector

$$\dot{\mathbf{p}}_d(\mathbf{p}_k) := E\nabla\Phi - k_e\Phi\nabla\Phi, \quad E = \begin{bmatrix} 0 & 1 \\ -1 & 0 \end{bmatrix}, \quad (3)$$

where E specifies the rotation (which influence the tracking direction), and $k_e \in \mathbb{R}_{\geq 0}$ the gain to adjust the speed of convergence. The direction the velocity vector $\dot{\mathbf{p}}_d$ is pointing at is generally different from the course heading due to the atmospheric interference, such as wind ($w \in \mathbb{R}$ in Figure 1).

C. Energy evolution due to trajectory

The algorithm models the energy using as state energy coefficients $\mathbf{q} \in \mathbb{R}^m$ derived from Fourier analysis (the size of the energy coefficients vector m is related to the order of a Fourier series) and decomposes the evolution in energy due to the trajectory and computations. This approach is adapted from our earlier work on computational energy analysis [22], [32], and energy estimation of a fixed-wing UAV [23].

Let us consider a periodic signal of period $T \in \mathbb{Z}_{>0}$ representing the energy consumption, and Fourier series of an arbitrary order $r \in \mathbb{Z}_{\geq 0}$ and for the purpose of energy consumption modeling of the signal

$$h(t) = a_0/T + (2/T) \sum_{j=1}^r (a_j \cos \omega j t + b_j \sin \omega j t), \quad (4)$$

where $h : \mathbb{R}_{\geq 0} \rightarrow \mathbb{R}$ maps time to the modeled instantaneous energy consumption, $\omega := 2\pi/T$ is the angular frequency, and $a, b \in \mathbb{R}$ the Fourier series coefficients.

Consider a linear time-invariant (LTI) state-space model

$$\begin{aligned} \dot{\mathbf{q}} &= A\mathbf{q} + B\mathbf{u} + \mathbf{w}, \\ y &= C\mathbf{q} + v, \end{aligned} \quad (5)$$

where $y \in \mathbb{R}_{\geq 0}$ is the instantaneous energy consump-

tion. The state \mathbf{q} are the energy coefficients

$$\begin{aligned} \mathbf{q} &= [\alpha_0 \quad \alpha_1 \quad \beta_1 \quad \cdots \quad \alpha_r \quad \beta_r]^T, \\ A &= \begin{bmatrix} 0 & & & & & \\ & A_1 & & & & \\ & & \ddots & & & \\ & & & A_r & & \end{bmatrix}, \quad A_j \begin{bmatrix} 0 & \omega j \\ -\omega j & 0 \end{bmatrix}, \\ C &= (1/T) [1 \quad 1 \quad 0 \quad \cdots \quad 1 \quad 0], \end{aligned} \quad (6)$$

where $\mathbf{q} \in \mathbb{R}^m$ given $m = 2r + 1$, $A \in \mathbb{R}^{m \times m}$ is the state transmission matrix, and $C \in \mathbb{R}^m$ is the output matrix. In matrix A , the top left entry is one, the diagonal entries are A_1, \dots, A_r , the remaining entries are zero.

Lemma II.1 (Signals equality). Suppose there is no state and output uncertainty (the contributions $\mathbf{w} \in \mathbb{R}^m, v \in \mathbb{R}$ are zero) and the control is zero. Suppose further matrices A, C are described by Equation (6), and the initial guess \mathbf{q}_0 is $\mathbf{q}_0 = [a_0 \quad a_1/2 \quad b_1/2 \quad \cdots \quad a_r/2 \quad b_r/2]$. Then Equation (4) describes the same signal as Equation (5).

Proof. The equivalence of the models is trivial and the equality of the two signals is achieved by a proper choice of items of matrices A, C and the initial guess \mathbf{q}_0 . We refer the reader to Appendix I for a formal proof of the equality. ■

Let us consider the discretized version A_d of the matrix A can be expressed $A_d = A + I$ for small enough values of time step (forward Euler approximation¹).

Suppose at time instant k the plan reached stage i and the control input is $\mathbf{u}_k^a := \langle \mathbf{c}_k, \mathbf{s}_k \rangle$. The control along with the input matrix

$$\begin{aligned} \mathbf{u}_k &= [g(\mathbf{s}_k) - g(\mathbf{s}_{k-1}) \quad \mathbf{c}_k - \mathbf{c}_{k-1}]^T, \\ B &= \begin{bmatrix} 1 & \omega_{i,1} & \cdots & \omega_{i,\rho} \\ & 0 & & \\ & & \ddots & \\ & & & 0 \end{bmatrix}, \end{aligned} \quad (7)$$

where $\mathbf{s}_k, g(\mathbf{s}_k)$ are the computations and the instantaneous computational energy consumption (Subsection II-D), $\mathbf{u}_k \in \mathbb{R}^n$ is the control given $n = 1 + \rho$, and $B \in \mathbb{R}^{m \times n}$. Moreover, the top-left entry of B is one, while the others on the first row are gain factors $\Omega_i := \{\omega_{i,1}, \dots, \omega_{i,\rho}\} \in \mathbb{R}^\rho$, quantifying the contribution of a given adjustment to the instantaneous energy consumption.

Equation (7) accounts for the energy due to the adjustments and computations. The difference $g(\mathbf{s}_k) - g(\mathbf{s}_{k-1})$ quantifies the change in the instantaneous computational

¹One can guarantee to have the same outputs rather than an approximation with $A = e^{A_c k}$

energy consumption. Similarly, $\Omega_i(\mathbf{c}_k - \mathbf{c}_{k-1})$ quantifies the change in the instantaneous trajectory energy consumption. For instance, when the TEE φ_0 is a circle (see Figure 1), a decrement in the adjustment radius of the circle $c_{0,1}$ adds a negative contribution. It thus simulates the lowering of instantaneous energy consumption $\omega_{0,1}(c_{0,1} - c_{0,1}^-) < 0$.

Note from Equation (7) that the control input \mathbf{u}_k^a differs from the nominal control \mathbf{u}_k in Equation (5). The first is an output of the algorithm. The second is a function that maps two consecutive control inputs to the difference in instantaneous energy consumption $\mathbf{u}_k : \mathbb{R}^{2\rho} \times \mathbb{Z}_{\geq 0}^{2\sigma} \rightarrow \mathbb{R}^{1+\rho}$.

The energy consumption modeling of the plan utilizes the following assumption.

Assumption II.2 (Energy evolution periodicity). Given two time instants k_1, k_2 s.t. $k_1 > k_2$ and a constant value $n \in \mathbb{Z}_{>0}$

$$|y_k - y_{k+n}| \in \mathbb{E} \subset \mathbb{R}_{\geq 0} \quad \forall k \in [k_1, k_2]. \quad (8)$$

Physically this means that the time evolution of the instantaneous energy consumption is assumed to be approximately periodic.

D. Energy evolution due to computations

The energy cost of the computations (the set of tasks being executed on an embedded board on the UAV) is assessed using `powprofiler`, an open-source modeling tool presented in our previous work [22]. The tool measures the instantaneous computational energy consumption of software components within the QoS sets. It builds an energy model: a linear interpolation, one per each task. In a ROS based system, it requires the user to implement one or more ROS nodes changing the computational load by node-specific ROS parameters. The approach has been tested in simulation in our previous work [33].

Specifically, the mission \mathcal{M} contains a set of ordered lists with tasks (recall Definition II.1). These tasks are implemented by software components (such as ROS nodes in a ROS based system) $\Psi(\mathbf{s}_i) := \langle \psi_1(s_{i,1}), \dots, \psi_\sigma(s_{i,\sigma}) \rangle$. They input the desired and output the actual computations

$$\mathbf{s}_i := \{ \langle s_{i,1}, \dots, s_{i,\sigma} \rangle \mid \psi_j(s_{i,j}) \in \mathbb{S}_i \forall j \in [\sigma]^+ \}, \quad (9)$$

where $s_{i,j} : \mathbb{Z}_{\geq 0} \rightarrow \mathbb{Z}_{\geq 0}$ returns the j -th desired computation at stage i , $\mathbf{s}_i \in \mathbb{S}_i \subseteq \mathbb{Z}_{\geq 0}^\sigma$ the union of all QoS sets given $\mathbb{S}_i := \bigcup_{j \in [\sigma]^+} \mathbb{S}_{i,j}$.

For instance, if the task ψ_1 is a CNN object detector, $s_{1,1}$ corresponds to the computation frames-per-second (fps) rate changing the detection frequency. The algorithm outputs the control input which contains the desired fps rate $s_{1,1} \in \mathbb{S}_{1,1}$. The task $\psi_1(s_{1,1})$ outputs the

actual fps rate $s_{k,1}$ which might differ from the desired one (e.g., the CNN object detector might not be able to reach a high fps rate with the current computational resources).

Let us further define $g : \mathbb{Z}_{\geq 0}^\sigma \rightarrow \mathbb{R}_{\geq 0}$ as the instantaneous computational energy consumption value obtained using `powprofiler`

$$y_k^s := g(\Psi(\mathbf{s}_i)) = g(\mathbf{s}_k). \quad (10)$$

Moreover, let $g(\{\emptyset\})$ be zero.

III. ALGORITHM

Given l stages \mathcal{M}_i (TEE φ_i , tasks Ψ , computations $\mathbf{s}_i \in \mathbb{S}_i$, and adaptations $\mathbf{c}_i \in \mathbb{C}_i$ for all $i \in [l]$), the main purpose of the algorithm is to output a control input sequence $\mathbf{u}^a := \{\mathbf{u}_0^a, \mathbf{u}_1^a, \dots\}$ in a valid mission.

Definition III.1 (Valid mission). A mission is valid if for every stage \mathcal{M}_{i-1} , $i \in [l]^+$ there exist a control input \mathbf{u}_k^a that produce the next stage \mathcal{M}_i

$$\begin{aligned} \mathbf{u}_k^a &= \{ \langle \mathbf{c}_k, \mathbf{s}_k \rangle \mid \exists n \in \mathbb{Z}_{>0}, \\ &\langle \varphi_{i-1}(\mathbf{p}_{k-n}, \mathbf{c}_{k-n}), \Psi(\mathbf{s}_{k-n}) \rangle \in \mathcal{M}_{i-1} \quad (11) \\ &\implies \langle \varphi_i(\mathbf{p}_k, \mathbf{c}_k), \Psi(\mathbf{s}_k) \rangle \in \mathcal{M}_i \}. \end{aligned}$$

Let us proof that if the mission is valid, the instantaneous energy consumption can be modeled as linear combination of the state from the Equation (5).

Theorem III.1 (Periodic energy model). Consider the mission from Definition II.1, the valid mission from III.1. Assume Assumption II.2 holds, the model of Equation (5) behaves ideally ($\mathbf{w} = \mathbf{0}, v = 0$), the initial energy coefficients state \mathbf{q}_0 is y_0^a/m for the first coefficient where $y_0^a \in \mathbb{R}_{>0}$ is an initial measurement², $(1/2)y_0^a/m$ for all the others, and the mission is valid. Then, the instantaneous energy consumption y_k is a linear combination of the state \mathbf{q}_k .

Proof. The proof is based on mathematical induction. Base case: we proof that $y_0 = y_0^a$. Recall the definition of the state in Equation (6). The output is $y_0 = \alpha_{0,0} + \alpha_{0,1} + \dots + \alpha_{0,r} = y_0^a/m + (1/2)y_0^a/m + \dots + (1/2)y_0^a/m = y_0^a$.

Induction step: by inspection of Equation (5), the output at instant k can be expressed $y_k = (\alpha_{0,0} + B\mathbf{u}_0 + \dots + B\mathbf{u}_{k-1}) + p_1(k)\alpha_{0,1} + \dots + p_r(k)\alpha_{0,r}$, where $\forall t \in \mathbb{Z}_{\geq 2}$

$$p_r(t) := \begin{cases} \prod_{i=1}^{t/2} r^3/\xi^3 & \text{for even } t \\ (r/\xi) \prod_{i=1}^{(t-1)/2} r^3/\xi^3 & \text{for odd } t \end{cases}. \quad (12)$$

Suppose k is even and the theorem holds up to k . Initial energy coefficients state \mathbf{q}_0 leads to $y_k =$

² y_0^a can be the initial measurement or the measurement from a previous instance of the algorithm

$$(y_0^a/m + B\mathbf{u}_0 + \dots + B\mathbf{u}_{k-1}) + p_1(k)(1/2)y_0^a/m + \dots + p_r(k)(1/2)y_0^a/m = y_k^a.$$

We prove now that the instantaneous energy consumption at $k+1$ is still a linear combination of the state. We express the output in function of the previous state $y_{k+1} = (\alpha_{0,0} + B\mathbf{u}_0 + \dots + B\mathbf{u}_k) + (1/\xi)\beta_{k,1} + \dots + (r/\xi)\beta_{k,r}$. Notice that the coefficients α, β have an equivalent evolution (indeed this allows to simulate the periodicity) and $\beta_{k,r} = p_r(k)\beta_{0,r}$. Thus, the output can be expressed $y_{k+1} = (\alpha_{0,0} + B\mathbf{u}_0 + \dots + B\mathbf{u}_k) + (1/\xi)p_1(k)\beta_{0,1} + \dots + (r/\xi)p_r(k)\beta_{0,r}$. The expression is equivalent to $y_{k+1} = (\alpha_{0,0} + B\mathbf{u}_0 + \dots + B\mathbf{u}_k) + p_r(k+1)\beta_{0,r} + \dots + p_r(k+1)\beta_{0,r}$ using the definition of p_r in Equation (12). Again, the state \mathbf{q}_0 leads to $y_{k+1} = (y_0^a/m + B\mathbf{u}_0 + \dots + B\mathbf{u}_k) + p_r(k+1)(1/2)y_0^a/m + \dots + p_r(k+1)(1/2)y_0^a/m = y_{k+1}^a$, alike the previous statement, but at instant $k+1$. The proof for odd k is equivalent. ■

A. Output constraints set

We stated earlier the output y_k —the instantaneous energy consumption—evolves in $\mathbb{R}_{\geq 0}$. This is generally untrue. Physical UAVs are bounded by strict energy budgets due to battery limitations.

Let us hence consider the state of charge (SoC) of such battery with a simplistic difference equation [23]

$$\text{SoC}_k = - \left(V - \sqrt{V^2 - 4R_r\tilde{V}y_kV^{-1}} \right) / 2R_rQ_c, \quad (13)$$

where $V \in \mathbb{R}$ is the internal battery and $\tilde{V} \in \mathbb{R}$ the stabilized voltage, $R_r \in \mathbb{R}$ the resistance, and $Q_c \in \mathbb{R}$ the constant nominal capacity. We define the output constraints set

$$\mathbb{Y}_k := \{y_k \mid y_k \in [0, \text{SoC}_kQ_cV] \subseteq \mathbb{R}_{\geq 0}\}, \quad (14)$$

and $\max \mathbb{Y}_k$ is the maximum discharge capacity by the internal battery voltage—the maximum instantaneous energy consumption.

B. Deployment algorithm

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1: procedure STEP( $\mathbf{q}_{k-1}, \mathbf{u}_{k-1}^a, \mathbf{u}_{k-2}^a, P_{k-1}$ )
2:    $\mathbf{u}_{k-1} \leftarrow \mathbf{u}_{k-1}(\max \mathbf{u}_{k-1}^a, \mathbf{u}_{k-2}^a)$ 
3:    $\mathbf{u}_{k-1}^0 \leftarrow \arg \max_{\mathbf{u}} \sum_{i=k-1}^{k+N-2} l(\mathbf{q}_i, \mathbf{u}_i) + V_f(\mathbf{q}_{k+N-1})$ 
4:    $\hat{\mathbf{q}}_k^- \leftarrow A\hat{\mathbf{q}}_{k-1} + B\mathbf{u}_{k-1}^0$ 
5:   if  $C\hat{\mathbf{q}}_k^- \notin \mathbb{Y}_k$  then
6:      $\mathbf{u}_{k-1}^a \leftarrow \mathbf{u}_{k-1}^a / \{\max \mathbf{u}_{k-1}^a\}$ 
7:     return STEP( $\mathbf{q}_{k-1}, \mathbf{u}_{k-1}^a, \mathbf{u}_{k-2}^a, P_{k-1}$ )
8:   else
9:     if  $|y_k^a - C\hat{\mathbf{q}}_k^-| \leq \varepsilon$  then
10:       $\hat{\mathbf{q}}_k \leftarrow \hat{\mathbf{q}}_k^-$ 
11:       $P_k \leftarrow P_{k-1}^-$ 
12:    else
13:       $P_k^- \leftarrow AP_{k-1}A^T + Q$ 

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14:    $K \leftarrow P_k^- C^T / (CP_k^- C^T + R)$ 
15:    $\hat{\mathbf{q}}_k \leftarrow \hat{\mathbf{q}}_k^- + K(y_k^a - C\hat{\mathbf{q}}_k^-)$ 
16:    $P_k \leftarrow (I + KC)P_k^-$ 
17:   end if
18:    $\mathbf{u}_k^a \leftarrow \mathbf{u}_{k-1}^a$ 
19:   return ( $\hat{\mathbf{q}}_k, P_k, \mathbf{u}_k^a$ )
20: end if
21: end procedure
22: procedure EADMPA( $\mathcal{M}, \mathbf{p}_0, \mathbf{q}_0$ )
23:    $k \leftarrow 0$ 
24:    $\mathbf{u}_{k-1}^a \leftarrow \{\emptyset\}$ 
25:    $\mathbf{p}_k \leftarrow \mathbf{p}_0$ 
26:    $\mathbf{q}_k \leftarrow \mathbf{q}_0$ 
27:   while  $k \leq t_f$  do
28:      $\mathbf{u}_k^a \leftarrow \{\mathbf{u}_k^a \mid (\varphi_i(\mathbf{p}_k, \mathbf{c}_i), \Psi(\mathbf{s}_i)) \in \lambda(\mathbf{p}_k)\}$ 
29:      $(\mathbf{q}_k, P_k, \mathbf{u}_k^a) \leftarrow \text{STEP}(\mathbf{q}_k, \mathbf{u}_k^a, \mathbf{u}_{k-1}^a, P_k)$ 
30:      $\mathbf{p}_k \leftarrow \mathbf{p}_k \mathbf{P}_d(\mathbf{p}_k)/v$ 
31:      $\mathbf{u}_{k-1}^a \leftarrow \mathbf{u}_k^a$ 
32:      $k \leftarrow k + 1$ 
33:   end while
34: end procedure

```

Per each time step k (the final time t_f is unknown), the algorithm updates the state—the position at line 30 and the energy coefficients at line 29—and the control input. Note that the position can be computed directly from Equation (3). If the velocity is $v \in \mathbb{R}_{\geq 0}$, and the starting point $\mathbf{p}_0, \mathbf{p}_{k+1} = \mathbf{p}_k \mathbf{P}_d(\mathbf{p}_k)/v$.

In detail, initial guess for $P_0 \in \mathbb{R}^{j \times j}$ is positive definite and derived empirically, for \mathbf{q}_0 the initial measurement is distributed to the coefficients (see Theorem III.1). Line 2 selects the maximum possible control from the current control input. Line 3 uses robust output feedback model predictive control (MPC) [34] to select the optimal control \mathbf{u}^0 for a given horizon $N \in \mathbb{Z}_{>0}$ from the cost function

$$l(\mathbf{q}_k, \mathbf{u}_k) := (1/2)(\mathbf{q}_k^T Q \mathbf{q}_k + \mathbf{u}_k^T R \mathbf{u}_k), \quad (15)$$

$$V_f(\mathbf{q}_k) := (1/2)(\mathbf{q}_k^T P_f \mathbf{q}_k),$$

where matrices $Q \in \mathbb{R}^{j \times j}, R \in \mathbb{R}^{l \times l}$ are positive definite.

Follows a check if the mission can finish without the eventuality of battery discharge (output constraints satisfaction) at line 5, with the control input being eventually updated and the process reiterated at line 7.

Before the next step, state estimator—the discrete-time Kalman filter [35] at lines 13–16—predicts the state \mathbf{q} if the modeled instantaneous energy consumption diverges from the sensor's value y_k^a more than a given $\varepsilon \in \mathbb{R}_{\geq 0}$, or sensor measurements are unavailable ($y_k^a = 0$).

IV. RESULTS

Some preliminary results are discussed. Further analysis and re-implementation of the functionality of the algorithm is needed.

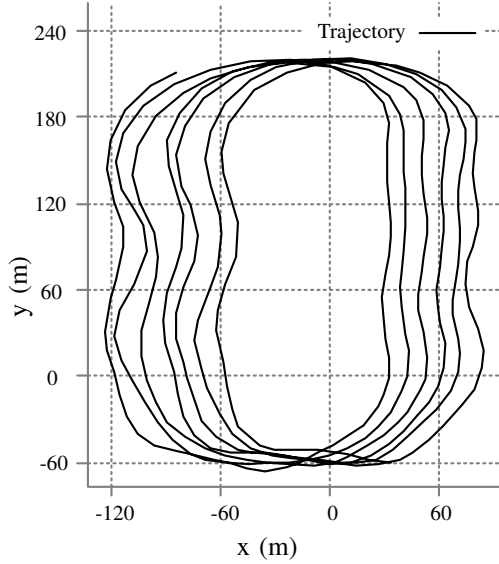


Fig. 4. The true trajectory. The trajectory is retrieved from the flight log.

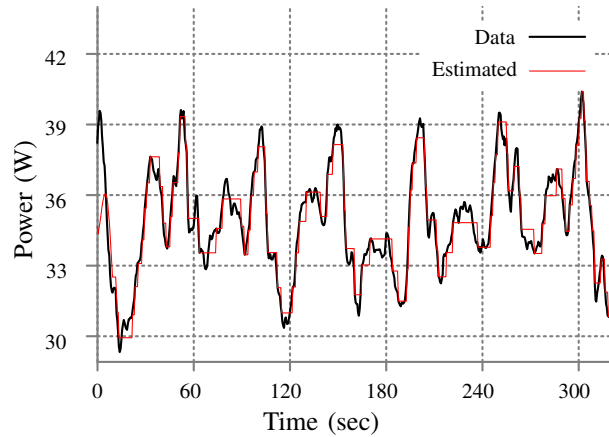


Fig. 5. Derived energy consumption from the sensors (throttle). Red line is the estimated value. We use KF for state estimation when the difference between the sensor and model value is $> \varepsilon$. Otherwise we evolve the state without noise.

V. CONCLUSION AND FUTURE WORK

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APPENDIX I
PROOF OF LEMMA II.1

We propose a formal proof of Lemma II.1. The proof justifies the choice of the items of the matrices A, C and of the initial guess \mathbf{q}_0 in Equation (6). We write these elements such that the coefficients of the series a_0, \dots, b_r are the same as the coefficients of the state α_0, \dots, β_r .

Let us re-write the Fourier series expression in Equation (4) in its complex form with the well-known Euler's formula $e^{it} = \cos t + i \sin t$. With $t = \omega j t$, we find the expression for $\cos \omega j t = (e^{i\omega j t} + e^{-i\omega j t})/2$ and $\sin \omega j t = (e^{i\omega j t} - e^{-i\omega j t})/(2i)$ by substitution of $\sin \omega j t$ and $\cos \omega j t$ respectively. This leads [36]

$$h(t) = a_0/T + (1/T) \sum_{j=1}^r e^{i\omega j t} (a_j - ib_j) + (1/T) \sum_{j=1}^r e^{-i\omega j t} (a_j + ib_j), \quad (16)$$

where i is the imaginary unit.

The solution at time t can be expressed $\mathbf{q} = e^{At} \mathbf{q}_0$. Both the solution and the system in Equation (5) are well established expressions derived using standard textbooks [36], [37]. To solve the matrix exponential e^{At} , we use the eigenvectors matrix decomposition method [38].

The method works on the similarity transformation $A = VDV^{-1}$. The power series definition of e^{At} implies $e^{At} = Ve^{Dt}V^{-1}$ [38]. We consider the non-singular matrix V , whose columns are eigenvectors of A ; $V := [v_0 \ v_1^0 \ v_1^1 \ \dots \ v_r^0 \ v_r^1]$. We then consider the diagonal matrix of eigenvalues $D = \text{diag}(\lambda_0, \lambda_1^0, \lambda_1^1, \dots, \lambda_r^0, \lambda_r^1)$. λ_0 is the eigenvalue associated to the first item of A . λ_j^0, λ_j^1 are the two eigenvalues associated with the block A_j . We can write $Av_j = \lambda_j v_j \ \forall j = \{1, \dots, m\}$, and $AV = VD$.

We apply the approach in terms of Equation (5), under the assumptions; $\dot{\mathbf{q}} = A\mathbf{q}$. The linear combination of the initial guess and the generic solution

$$F\mathbf{q}(0) = \gamma_0 v_0 + \sum_{k=0}^1 \sum_{j=1}^r \gamma_j v_j^k$$

$$F\mathbf{q}(t) = \gamma_0 e^{\lambda_0 t} v_0 + \sum_{k=0}^1 \sum_{j=1}^r \gamma_j e^{\lambda_j^k t} v_j^k$$

where $F = [1 \ \dots \ 1]$ is a properly sized vector of ones.

Let us consider the second expression. It represents the linear combination of all the coefficients of the state

at time t . It can also be expressed in the following form

$$F\mathbf{q}(t)/T = \gamma_0 e^{\lambda_0 t} v_0/T + (1/T) \sum_{j=1}^r \gamma_j e^{\lambda_j^0 t} v_j^0 + (1/T) \sum_{j=1}^r \gamma_j e^{\lambda_j^1 t} v_j^1. \quad (17)$$

We proof that the eigenvalues λ and eigenvectors V are such that Equation (17) is equivalent to Equation (16).

The matrix A is a block diagonal matrix, so we can express its determinant as the multiplication of the determinants of its blocks $\det(A) = \det(0) \times \det(A_1) \times \dots \times \det(A_r)$. We proof the first determinant and the others separately.

We proof that the first term of both the Equation (16) and (17) matches. We find the eigenvalue from $\det(0) = 0$, which is $\lambda_0 = 0$. The corresponding eigenvector can be chosen arbitrarily $(0 - \lambda_0)v_0 = [0 \ \dots \ 0] \ \forall v_0$, thus we choose $v_0 = [1 \ 0 \ \dots \ 0]$. We find the value γ_0 of the vector γ so that the terms are equal $\gamma_0 = [a_0 \ 0 \ \dots \ 0]$.

Then, we proof that all the terms in the sum of both the Equations (16) and (17) match.

For the first block A_1 , we find the eigenvalues from $\det(A_1 - \lambda I) = 0$. The polynomial $\lambda^2 + \omega^2$, gives two complex roots—the two eigenvalues $\lambda_1^0 = i\omega$ and $\lambda_1^1 = -i\omega$. The eigenvector associated with the eigenvalue λ_1^0 is $v_1^0 = [0 \ -i \ 1 \ 0 \ \dots \ 0]^T$. The eigenvector associated with the eigenvalue λ_1^1 is $v_1^1 = [0 \ i \ 1 \ 0 \ \dots \ 0]^T$. Again, we find the values γ_1 of the vector γ such that the equivalences

$$\begin{cases} e^{i\omega t} (a_1 - ib_1) &= \gamma_1 e^{i\omega t} v_1^0 \\ e^{-i\omega t} (a_1 + ib_1) &= \gamma_1 e^{i\omega t} v_1^1 \end{cases}$$

hold. They hold for $\gamma_1 = [b_1 \ a_1]$. The proof for the remaining $r - 1$ blocks is equivalent.

The initial guess is build such that the sum of the coefficients is the same in both the signals. The output matrix is built as follows. The coefficient $1/T$ accounts for the period in Equations (16–17) and (4). At time instant zero, the coefficients b_j are not present (and the coefficients a_j are doubled for each $j = 1, 2, \dots, r$). To match the outputs $h(t) = y(t)$ —or equivalently $F\mathbf{q}(t)/T = C\mathbf{q}(t)$ —we define $C = 1/T [1 \ 1 \ 0 \ \dots \ 1 \ 0]$. We thus conclude that the lemma holds.

We note that the signal would still be periodic with another linear combination of coefficients (for instance, $C = d [1 \ 0 \ 1 \ \dots \ 0 \ 1]$, or $d [1 \ \dots \ 1]$ for a constant value $d \in \mathbb{R}$). Even if both the signals h and y would be still periodic, they would not be equal. ■

APPENDIX II

PLAN SPECIFICATION EXAMPLE

We discuss how to specify an initial mission plan with an example of a survey mission. The reader can later execute the initial mission plan using the simulator in Appendix IV. The plan is user-defined. It specifies the trajectory and tasks of a mission along the QoS and TEEs sets.

Recall that the mission is composed of a set of stages (Section II). The plan has a variable number of entries per each stage. We plan the mission in Figures 1, 4. The first line

1: **0.0003; 90; (x+45)^2+(y-146)^2-4900+c1** corresponds to TEE $\varphi_0(\mathbf{p}_k, c_{0,1}) := (x + 45)^2 + (y - 146)^2 - 4900 + c_{0,1}$ in the initial stage \mathcal{M}_0 . The TEE is a circle. The gain is $k_e = 3 \cdot 10^{-4}$, and the rotation is of ninety degrees—we use the rotation matrix E from Equation (3). Note that line 1 further specifies the adjustment **c1** of the radius of the circle. It corresponds to $c_{0,1}$ in Definition II.1 (zero numerates the stage, one the adjustment).

The following line

2: **-sqrt(4900+c1)-45, 146**

specifies the triggering point. **sqrt** is the square root. The UAV reaches the triggering point within a given ε (the tolerance), and the algorithm switches to the next stage \mathcal{M}_1 . The algorithm ensures that the overall mission is still valid—Equation (11). It ensure that the TEEs and QoS constraints sets are satisfied with the generated control input. Note that the triggering point is in function of the adjustment **c1**. This is necessary; an adjustment in one stage affects the overall flight.

Let us assume the mission has two tasks. One task is the CNN object detection algorithm. Its computation $s_{0,1}$ is the fps rate. Another task is the variable key-size encryption algorithm. Its computation $s_{0,2}$ is the key-size. Then in mission plan

3: **[-3*10^3, 0]**

4: **[2, 10]**

5: **[32, 448]**

line 3 corresponds to TEE φ_0 's set $\mathbb{C}_0 = [-3 \cdot 10^3, 0]$. The algorithm selects $c_{0,1} \in \mathbb{C}_0$. Line 4 corresponds to the QoS set $\mathbb{S}_{0,1}$ for the computation $s_{0,1}$. Line 5 corresponds to the QoS set $\mathbb{S}_{0,2}$ for the computation $s_{0,2}$. Lines 3–5 must follow the ascending order of adjustments and computations. Lines 1–5 all describe the stage \mathcal{M}_0 .

Once the UAV reaches the triggering point (line 2) and there are no further lines, the algorithm terminates. If there are further lines, the stage switches to \mathcal{M}_1

6: **0.05; 270; x+sqrt(4900+c1)+45**

7: **-sqrt(4900+c1)-45, 11**

8: **[-3*10^3, 0]**

9: **[2, 10]**

10: **[32, 448]**

line 6 corresponds to TEE $\varphi_1(\mathbf{p}_k, c_{0,1}) := x + \sqrt{4900 + c_{0,1}} + 45$, a linear equation that intersects the triggering point (line 2). The gain is $k_e = 5 \cdot 10^{-2}$, and the rotation is opposite to line 1. The algorithm computes $-E$ from the rotation matrix in Equation 3.

The gain is different from the one used in φ_0 as different equations require different convergence rate. When we follow a circle it is useful to turn the rotation direction in advance. When we follow a straight line, it is preferable to turn the rotation direction closer to the line. We refer to the simulator in Appendix IV for more details on how to tune the gain k_e . The reader can simulate different rates to see how they affect the flight. The rotation is likewise different. In fact, the rotation matrix E points upwards in the space. It guides the UAV in the counter-clockwise direction (see Figure 1; the UAV is traveling counter-clockwise). The rotation matrix $-E$ points downwards and guides the UAV in the clockwise direction.

The triggering point of the stage \mathcal{M}_1 at line 7 also depends on the adjustment **c1**. Lines 8–10 specifies the constraints sets. They are expressed the same way as lines 3–5. They have to be specified explicitly for each stage, although they don't change. The current implementation of the algorithm has no other means to distinguish different constraints sets.

Again, if there are further lines and UAV reaches the triggering point (line 7), the algorithm switches the stage to \mathcal{M}_2

11: **0.0003; 90; (x+sqrt(4900+c1)-30)^2 + (y-11)^2-5625**

12: **-sqrt(4900+c1)+105, 11**

13: **[-3*10^3, 0]**

14: **[2, 10]**

15: **[32, 448]**

line 11 corresponds to TEE $\varphi_2(\mathbf{p}_k, c_{0,1}) := (x + \sqrt{4900 + c_{0,1}} - 30)^2 + (y - 11)^2 - 5625$, a circle of fixed size that changes the coordinates in function of the adjustment **c1**. The rotation is counter-clockwise, alike φ_0 .

The lines

16: **0.05; 90; x+sqrt(4900+c1)-105**

17: **-sqrt(4900+c1)+105, 147**

18: **[-3*10^3, 0]**

19: **[2, 10]**

20: **[32, 448]**

21: **0.0003; 90; (x-sqrt(4900+c1)+105)^2 + (y-147)^2-4900+c1**

22: **-3*sqrt(4900+c1)+105, 147**

23: **[-3*10^3, 0]**

defines the stages \mathcal{M}_3 and \mathcal{M}_4 . Line 16 corresponds

to TEE $\varphi_3(\mathbf{p}_k, c_{0,1}) := x + \sqrt{4900 + c_{0,1}} - 105$. Line 21 corresponds to TEE $\varphi_4(\mathbf{p}_k, c_{0,1}) := (x - \sqrt{4900 + c_{0,1}} + 105)^2 + (y - 147)^2 - 4900 + c_{0,1}$.

Note that the survey mission example changes the radius of the circular TEE φ_0 through the parameter $c_{0,1}$. It also changes the radius of φ_4 in the same way. The other TEEs ($\varphi_1, \varphi_2, \varphi_3$) displacement alters the distance between the lines in the survey. A familiar pattern from Figure 1. For simplicity, we suppose line 22 describes the last point of the mission. We further suppose the user does not desire to process any task in the last stage \mathcal{M}_4 . To inhibit the computations

24: **[0, 0]**

25: **[0, 0]**

If there are any following stages in the survey, they are constructed the same way. The stages \mathcal{M}_i for $i := \{0, 4, 8, \dots\}$ contain TEE circle with variable radius. For $i := \{1, 3, 5, \dots\}$ TEE line with variable displacement. For $i := \{2, 6, 10, \dots\}$ TEE circle with fixed radius and variable displacement.

We need a further abstraction before we can simulate

a mission in Appendix IV. The algorithm needs to know how to model the tasks—the software components.

APPENDIX III

COMPONENTS SPECIFICATION FORMAT

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APPENDIX IV

SIMULATOR

We discuss how to execute the initial mission plan and the component specification in the simulator. The reader can use the simulator to simulate different environmental interference and their effects on the mission.

The simulator is written in MATLAB and released under MIT license. It implements the algorithm (Subsection III) providing a position \mathbf{p}_k at each time step k and an energy sensor's value y_k^a . It is derived from empirical data. It is an abstraction of the physical observations.

The user is prompted to initialize the trajectory and the algorithm.