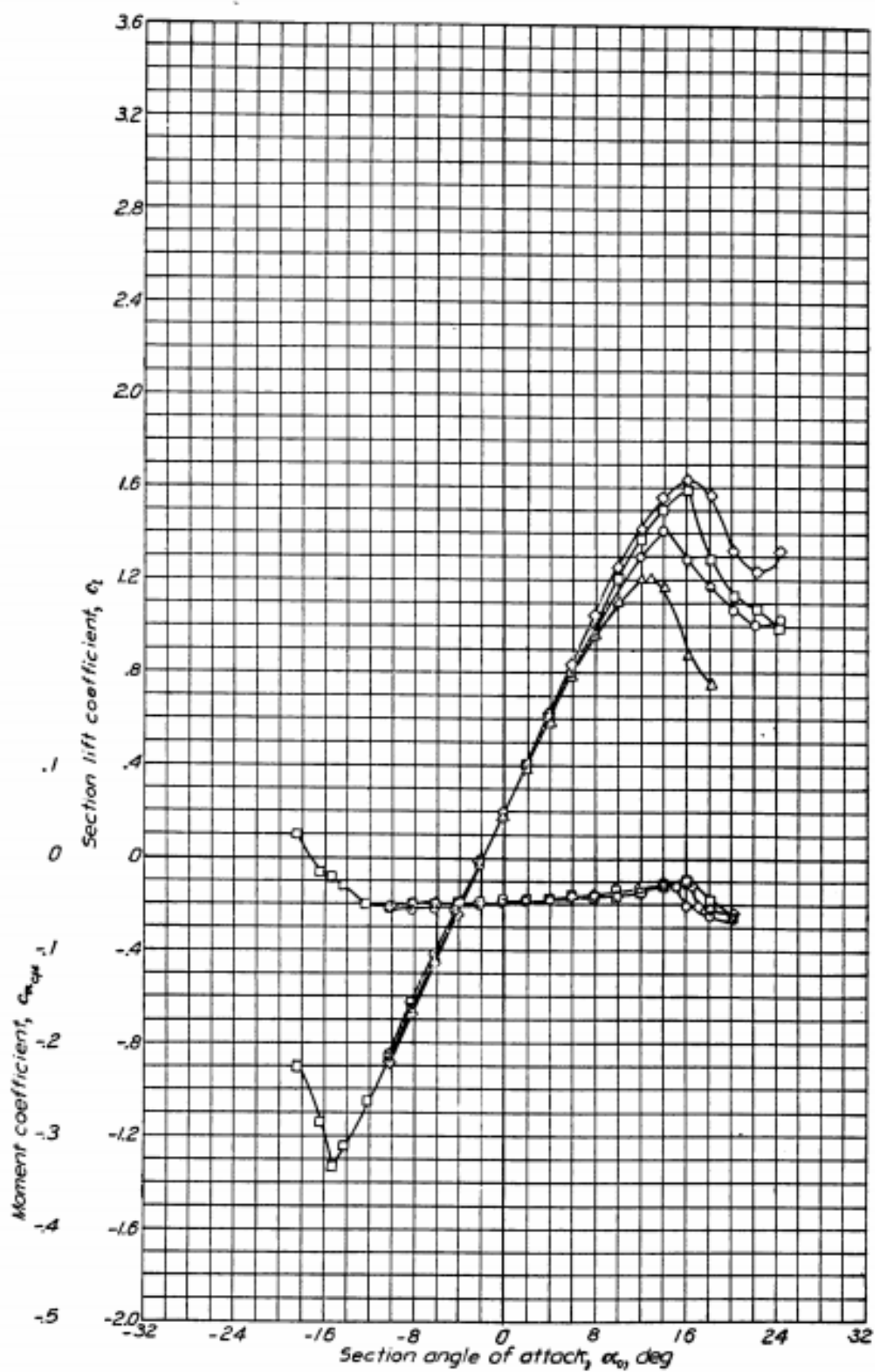


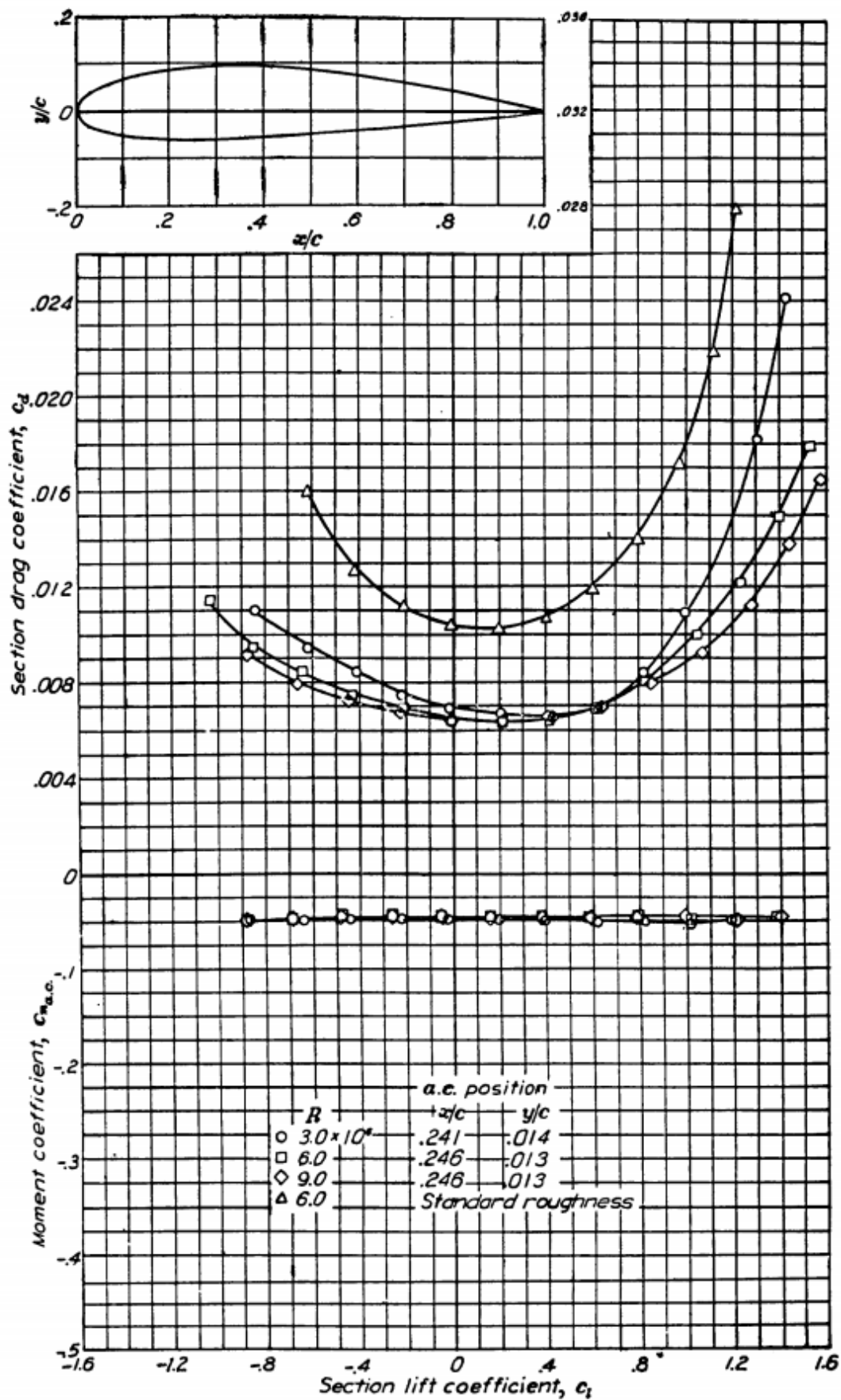
Homework Problems

1. Find the following coefficients at zero angles of attack for given aerofoils (NACA 2415, NACA 0009, NACA 23015, NACA 65₂-415)-
 - a) Lift coefficient (C_{l0})
 - b) Drag coefficient (C_{d0})
 - c) Pitching moment coefficient (C_{m0})
2. Find the lift curve slope ($C_{l\alpha}$), minimum drag coefficient ($C_{d\min}$), and pitching moment at the quarter chord of aerofoil from leading-edge ($C_{m\frac{c}{4}}$) for the linear region for given aerofoils (NACA 2415, NACA 0009, NACA 23015, NACA 65₂-415).

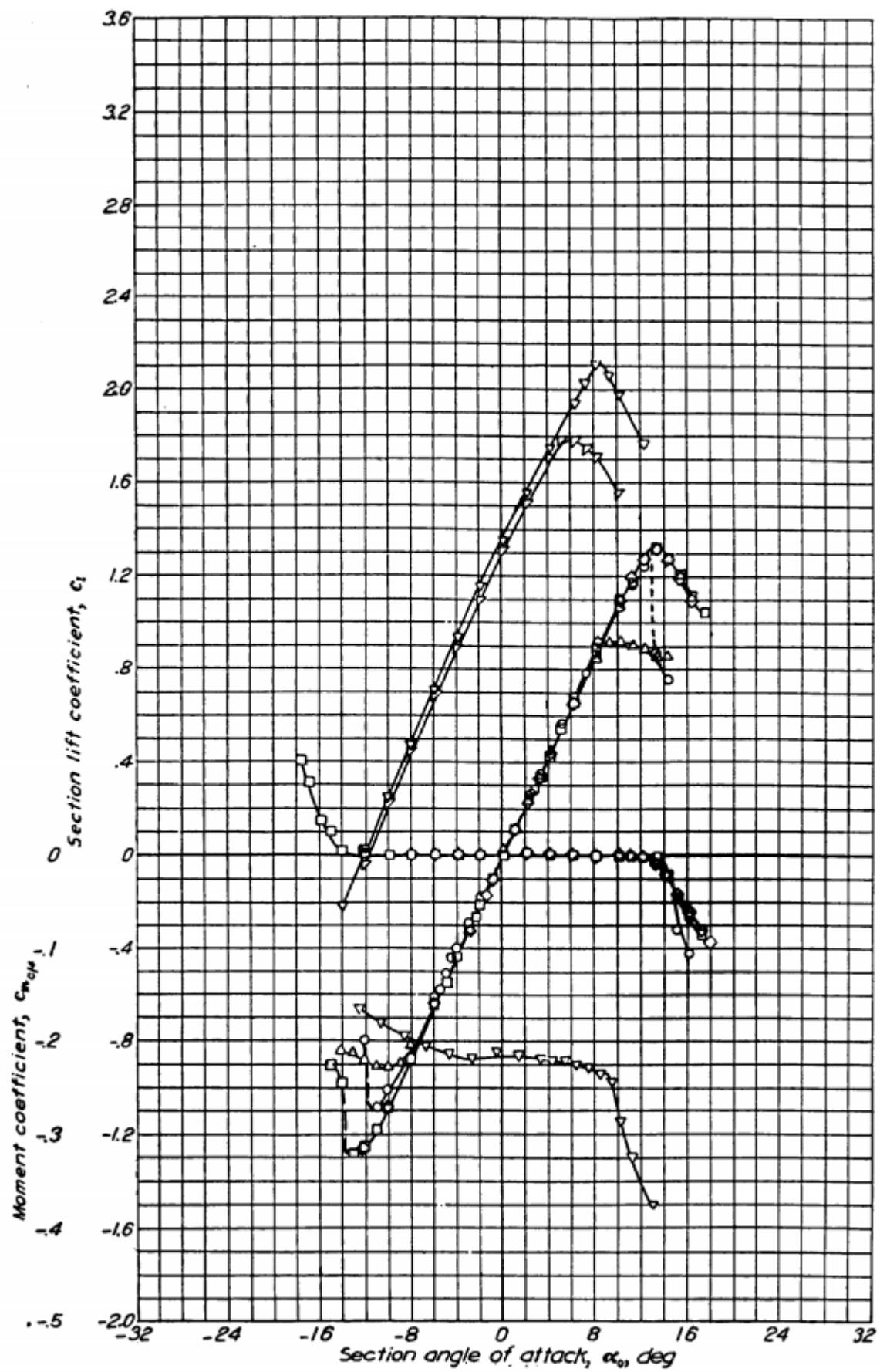
All the aerofoils data are given in the next pages.



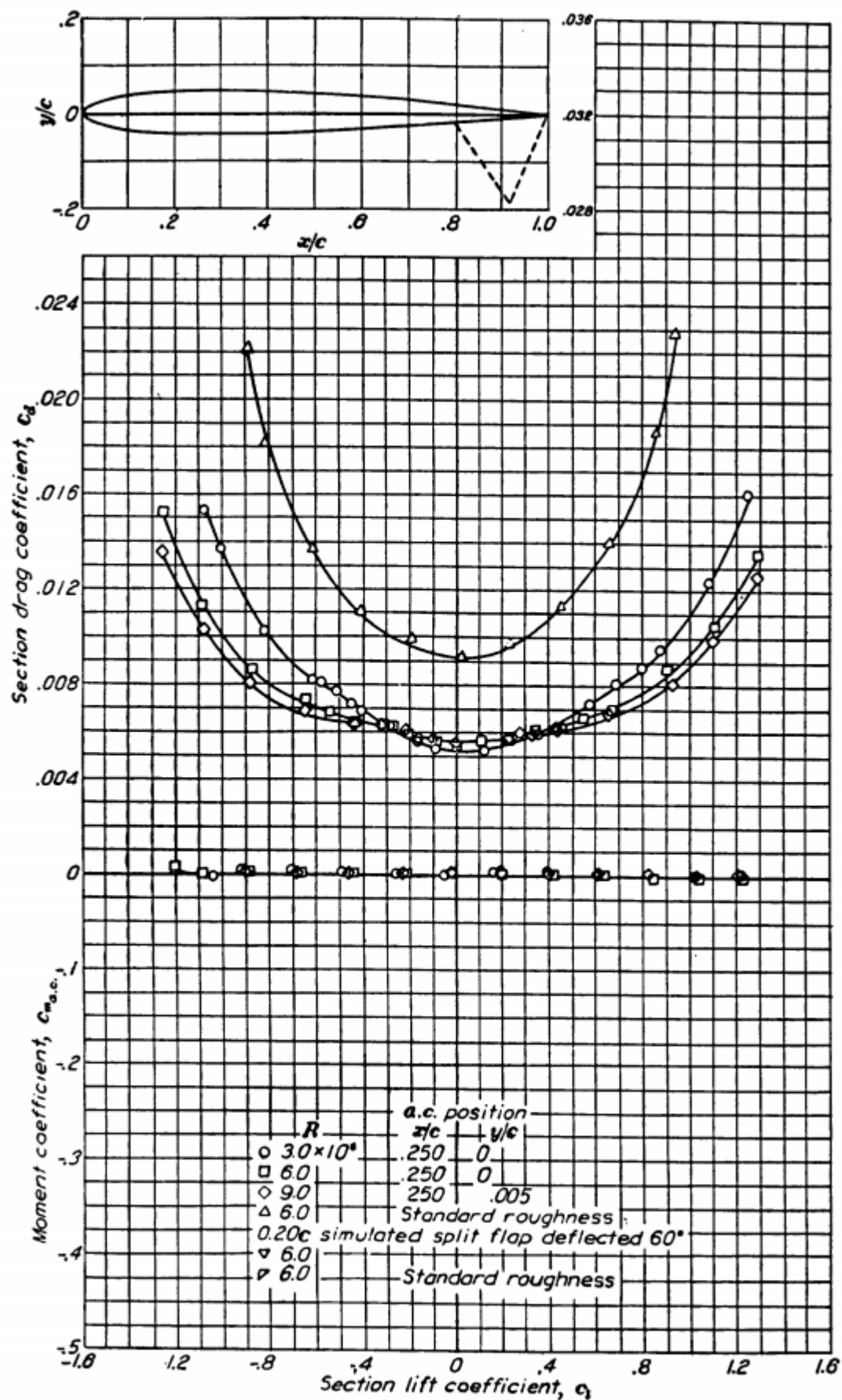
NACA 2415 Wing Section



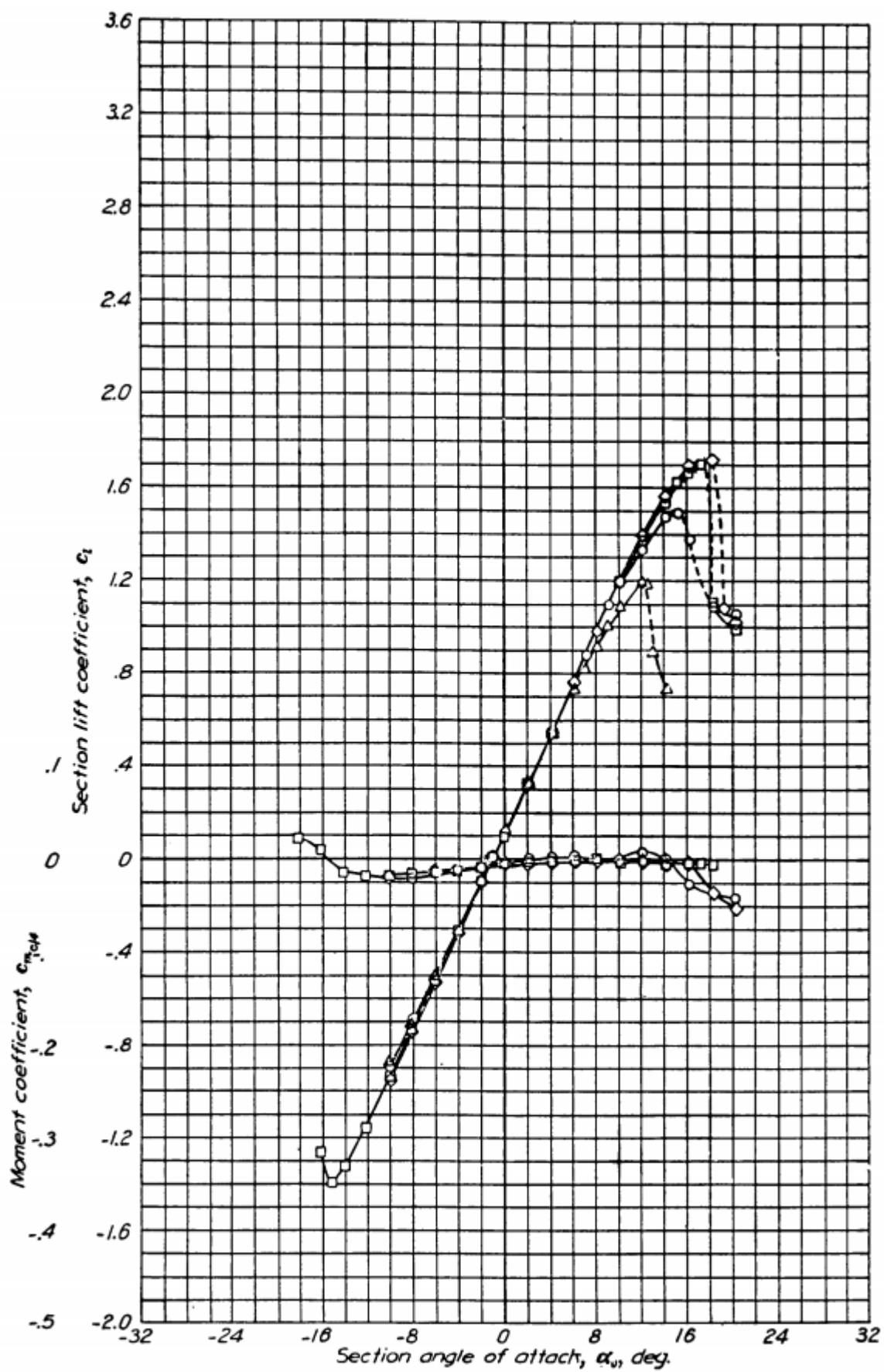
NACA 2415 Wing Section (Continued)



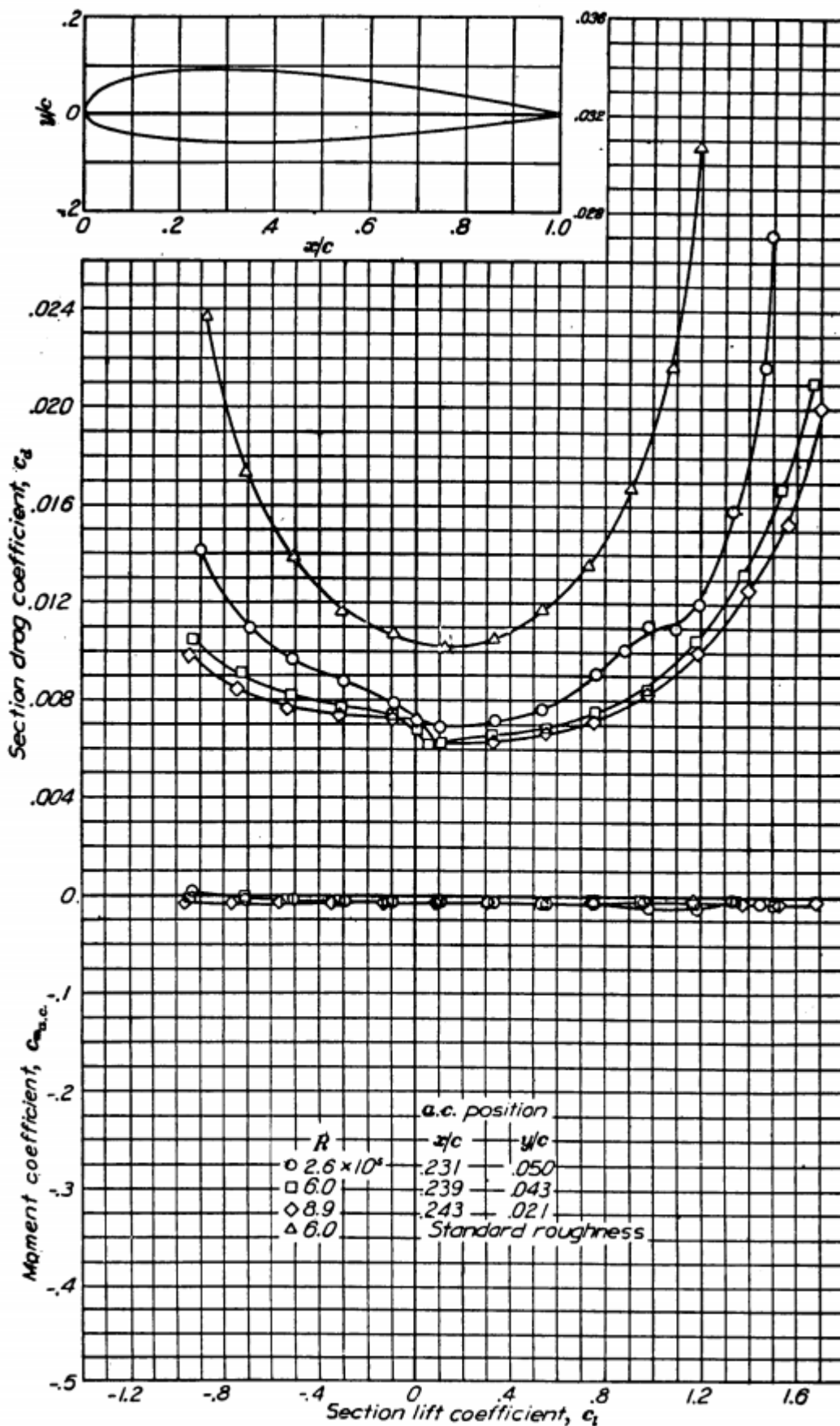
NACA 0009 Wing Section



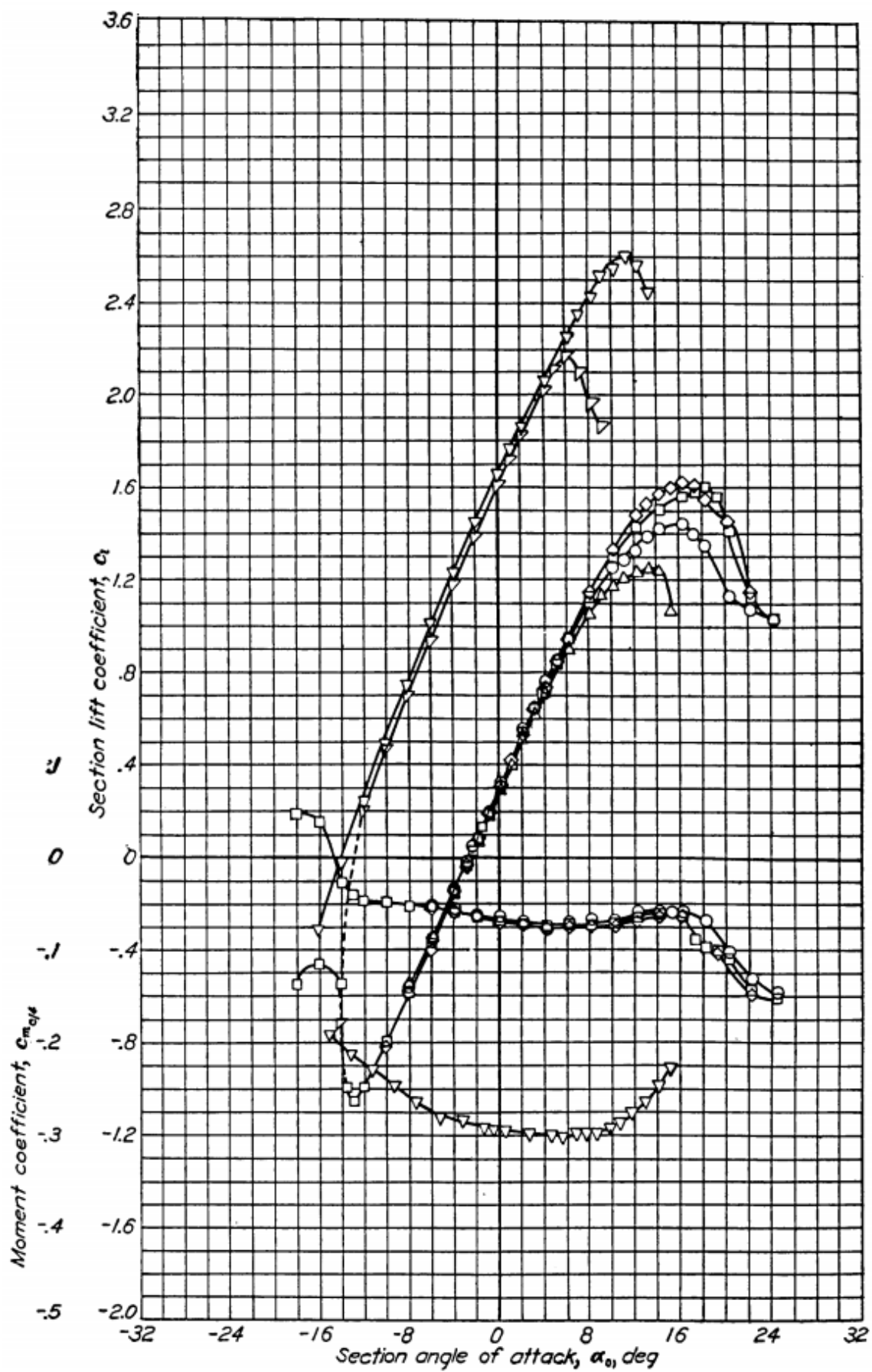
NACA 0009 Wing Section (Continued)



NACA 23015 Wing Section



NACA 23015 Wing Section (Continued)



NACA 65-415 Wing Section

