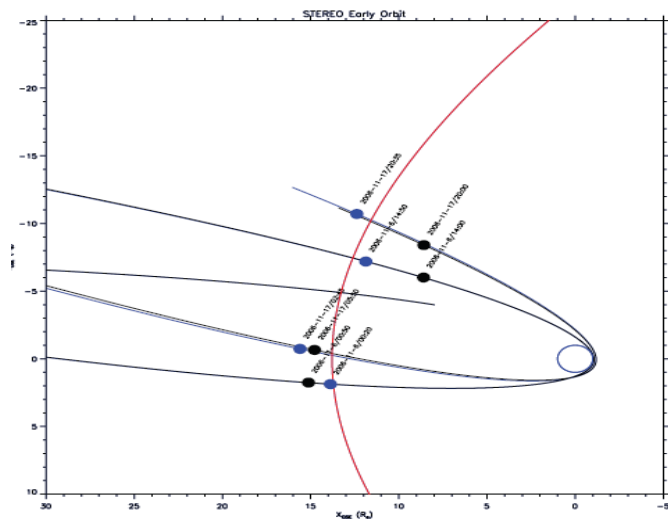
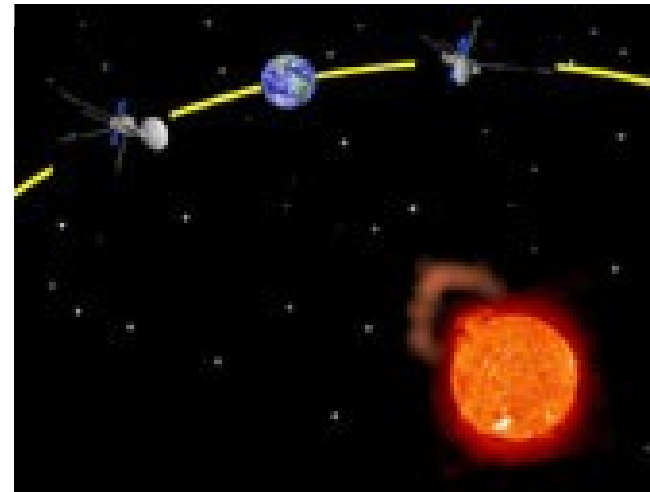
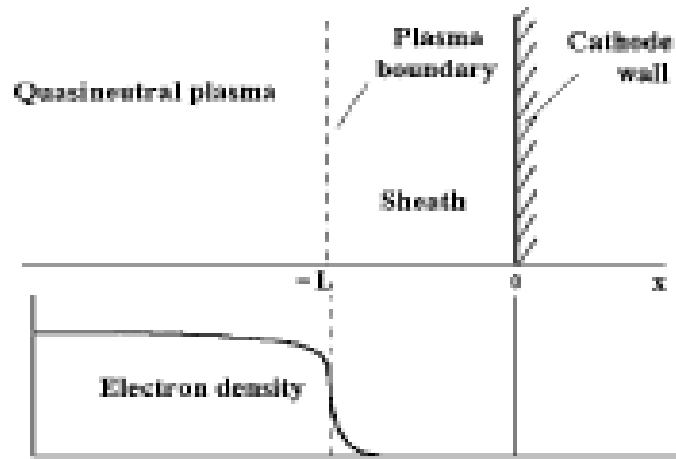
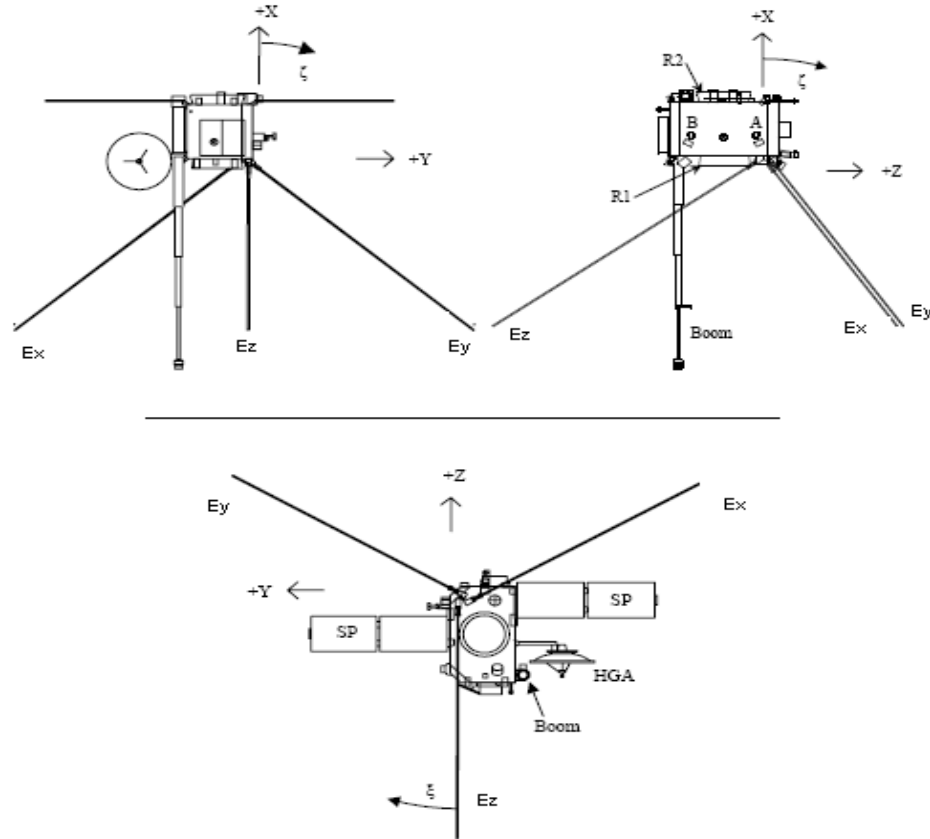


# The STEREO/WAVES antenna calibration

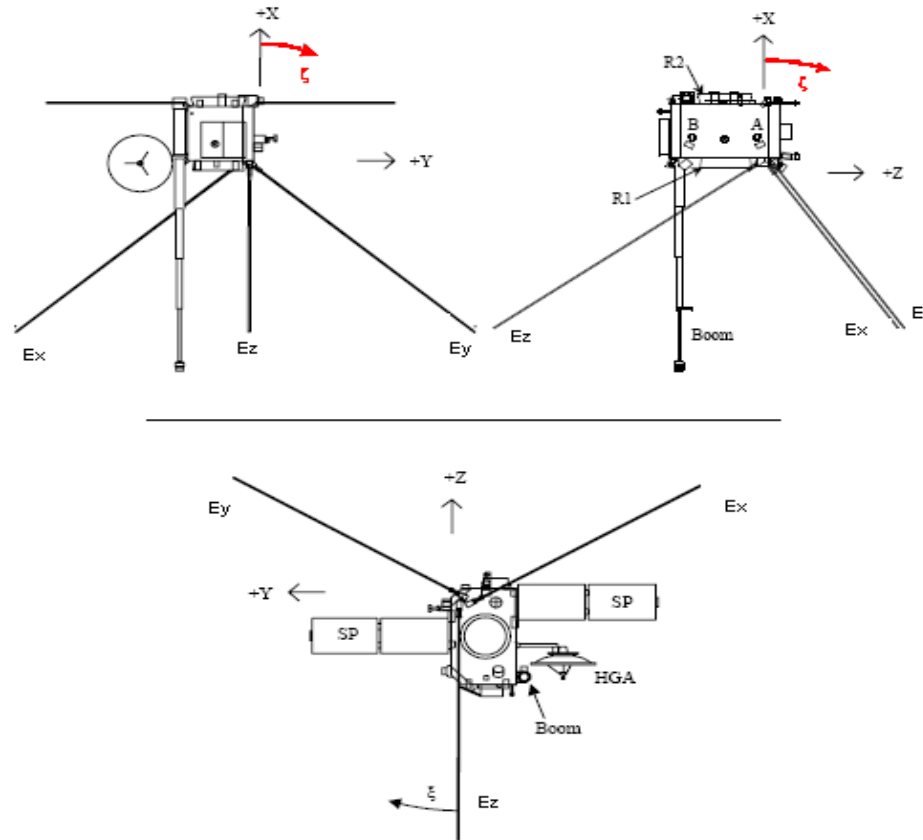


- The goal: correct data interpretation
- Effective length vectors, impedance/ admittance matrices, field patterns
- Numerical and experimental analysis:
  - Wire/Patchgrid modeling
  - Rheometry
  - Anechoic chamber
  - Inflight calibration

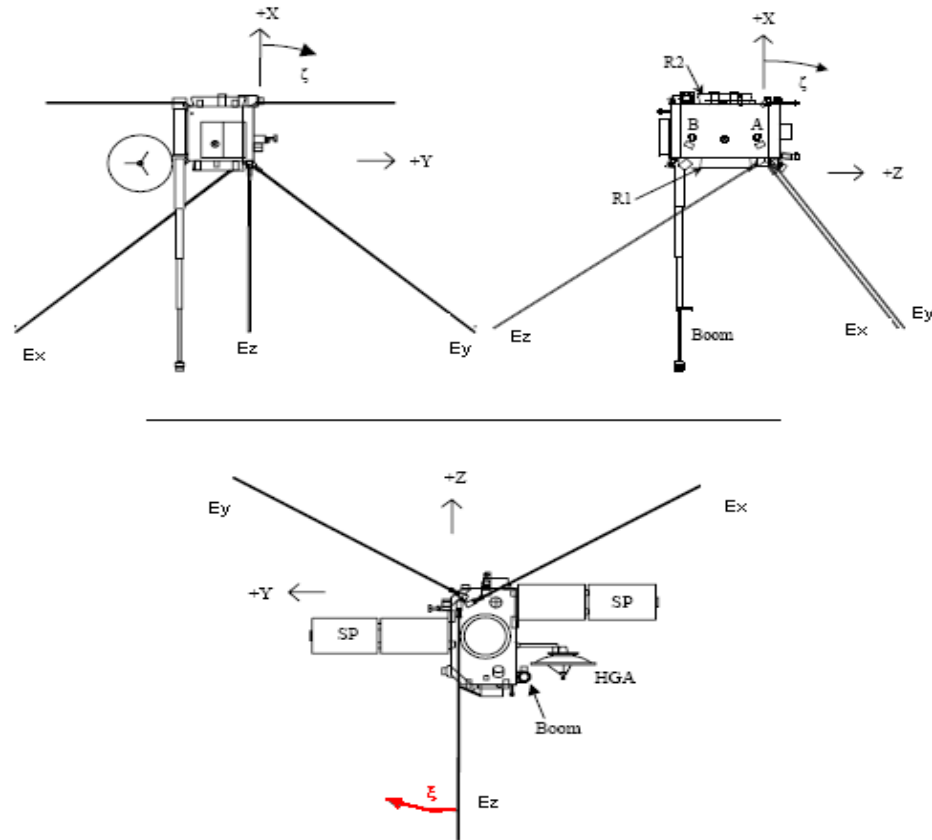




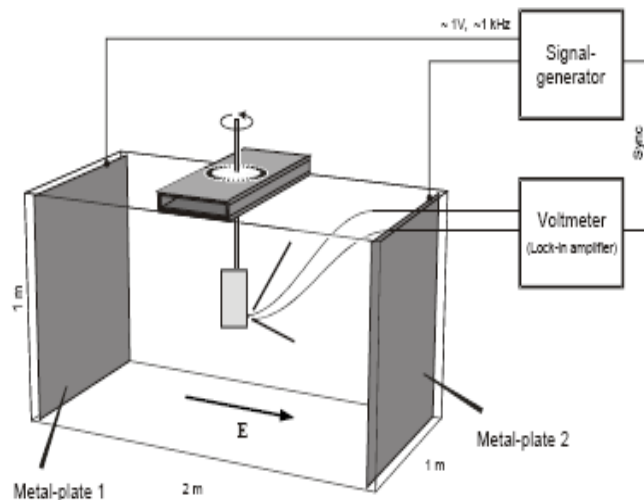
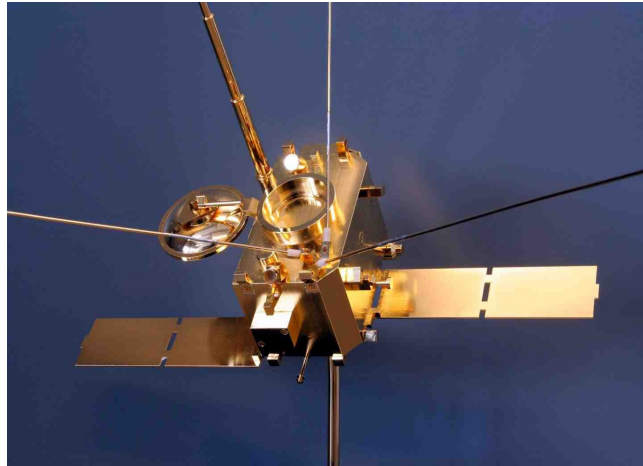
Antenna	$h^m$ [m]	$\zeta^m$ [deg]	$\xi^m$ [deg]
$E_x$	6.00	125.3	-120.0
$E_y$	6.00	125.3	120.0
$E_z$	6.00	125.3	0.0



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- A gold-plated model of the spacecraft is submerged into a water tank
- A low-frequency electric field is applied
- The response (induced voltage) of the antennas is measured as a function of spacecraft orientation
- The effective length vectors and the antenna impedances can be computed from the data
- Rheometry is only applicable to the quasi-static limit

HGA orientation	Antenna	STEREO A			STEREO B		
		$h^\circ$ [m]	$\zeta^\circ$ [deg]	$\xi^\circ$ [deg]	$h^\circ$ [m]	$\zeta^\circ$ [deg]	$\xi^\circ$ [deg]
-90 deg	$E_x$	2.89	126.5	-140.2	2.93	126.2	-140.8
	$E_y$	3.83	118.9	127.8	3.86	118.8	127.7
	$E_z$	2.37	132.2	21.1	2.36	132.4	20.0
0 deg	$E_x$	2.89	126.2	-140.7	2.92	126.0	-141.2
	$E_y$	3.84	118.7	127.9	3.87	118.8	127.6
	$E_z$	2.36	132.2	21.6	2.36	132.8	20.6
+90 deg	$E_x$	2.85	126.2	-140.8	2.89	125.8	-141.3
	$E_y$	3.84	118.6	128.1	3.86	118.6	127.2
	$E_z$	2.36	131.7	21.6	2.36	132.5	20.8

Antenna	$h^m$ [m]	$\zeta^m$ [deg]	$\xi^m$ [deg]
$E_x$	6.00	125.3	-120.0
$E_y$	6.00	125.3	120.0
$E_z$	6.00	125.3	0.0

open feeds

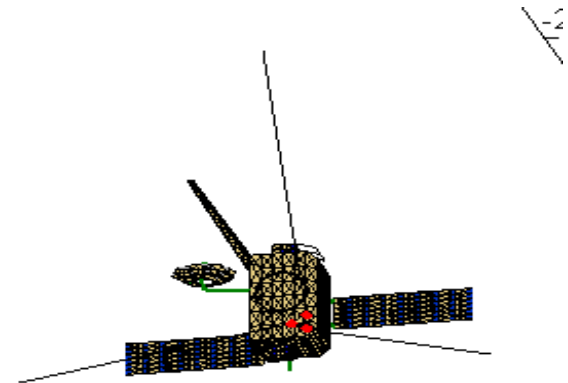
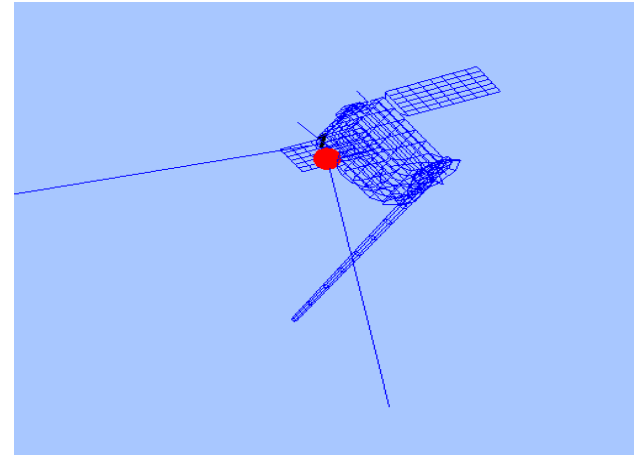
loaded feeds

HGA orientation n	Antenna	STEREO A			STEREO B		
		$h^\circ$ [m]	$\zeta^\circ$ [deg]	$\xi^\circ$ [deg]	$h^\circ$ [m]	$\zeta^\circ$ [deg]	$\xi^\circ$ [deg]
-90 deg	$E_x$	2.89	126.5	-140.2	2.93	126.2	-140.8
	$E_y$	3.83	118.9	127.8	3.86	118.8	127.7
	$E_z$	2.37	132.2	21.1	2.36	132.4	20.0
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	$E_y$	3.84	118.6	128.1	3.86	118.6	127.2
	$E_z$	2.36	131.7	21.6	2.36	132.5	20.8
		$h$ [m]	$\zeta$ [deg]	$\xi$ [deg]	$h$ [m]	$\zeta$ [deg]	$\xi$ [deg]
-90 deg	$E_x$	1.17	121.0	-134.5	1.17	120.8	-135.0
	$E_y$	1.46	114.9	126.6	1.46	114.8	126.4
	$E_z$	0.99	124.6	15.5	0.98	124.6	14.4
0 deg	$E_x$	1.16	120.8	-134.9	1.18	120.6	-135.4
	$E_y$	1.46	114.7	126.7	1.47	114.8	126.3
	$E_z$	0.99	124.6	15.9	0.98	125.0	14.9
+90 deg	$E_x$	1.15	120.8	-135.0	1.16	120.5	-135.5
	$E_y$	1.45	114.7	127.0	1.47	114.7	126.0
	$E_z$	0.98	124.3	15.9	0.98	124.9	15.0

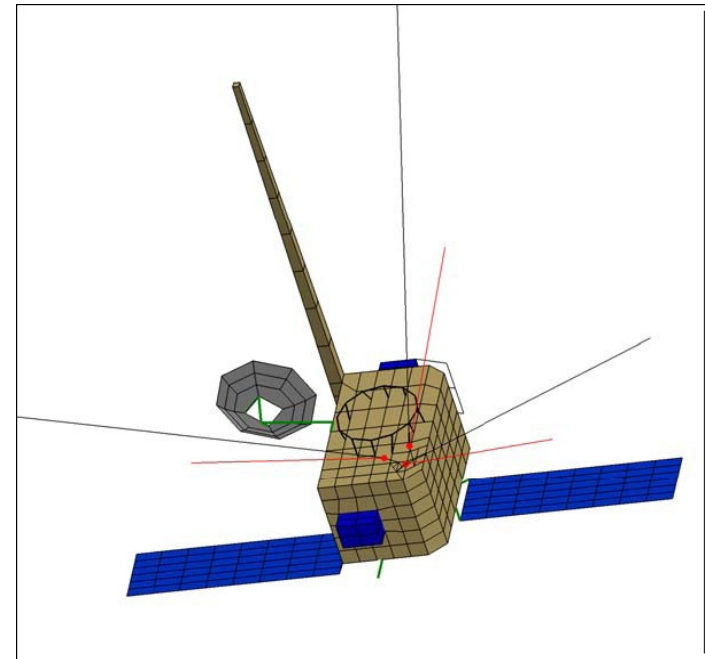


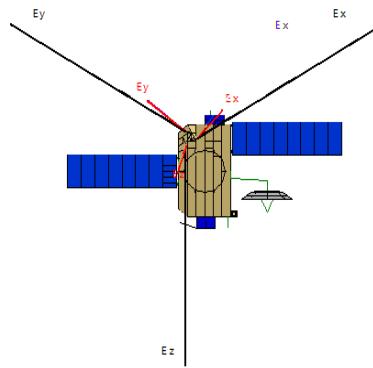
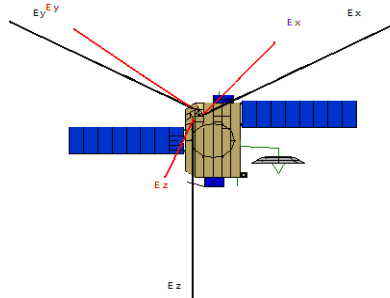
# The STEREO/WAVES antenna calibration

- The spacecraft is modeled as a grid of wires or patches
- Then the currents along these wires/patches are computed
- This calculation is done with ASAP (wires) and CONCEPT II (wires and patches)
- On base of the current distribution, all other antenna properties (effective length vectors, impedances) can be calculated with MATLAB routines



- The equation governing the current distribution is the electric field integral equation (EFIE, CONCEPT), or the reaction integral equation (RIE, ASAP)
- For patches, the magnetic field integral equation (MFIE, CONCEPT) is used
- The antenna is excited at the feed
- Due to reciprocity, the receiving antenna results in the same current distribution as the transmitting

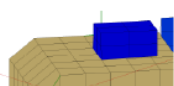
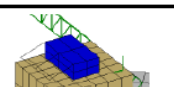
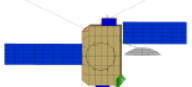
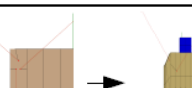
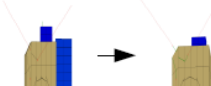
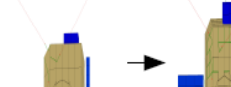





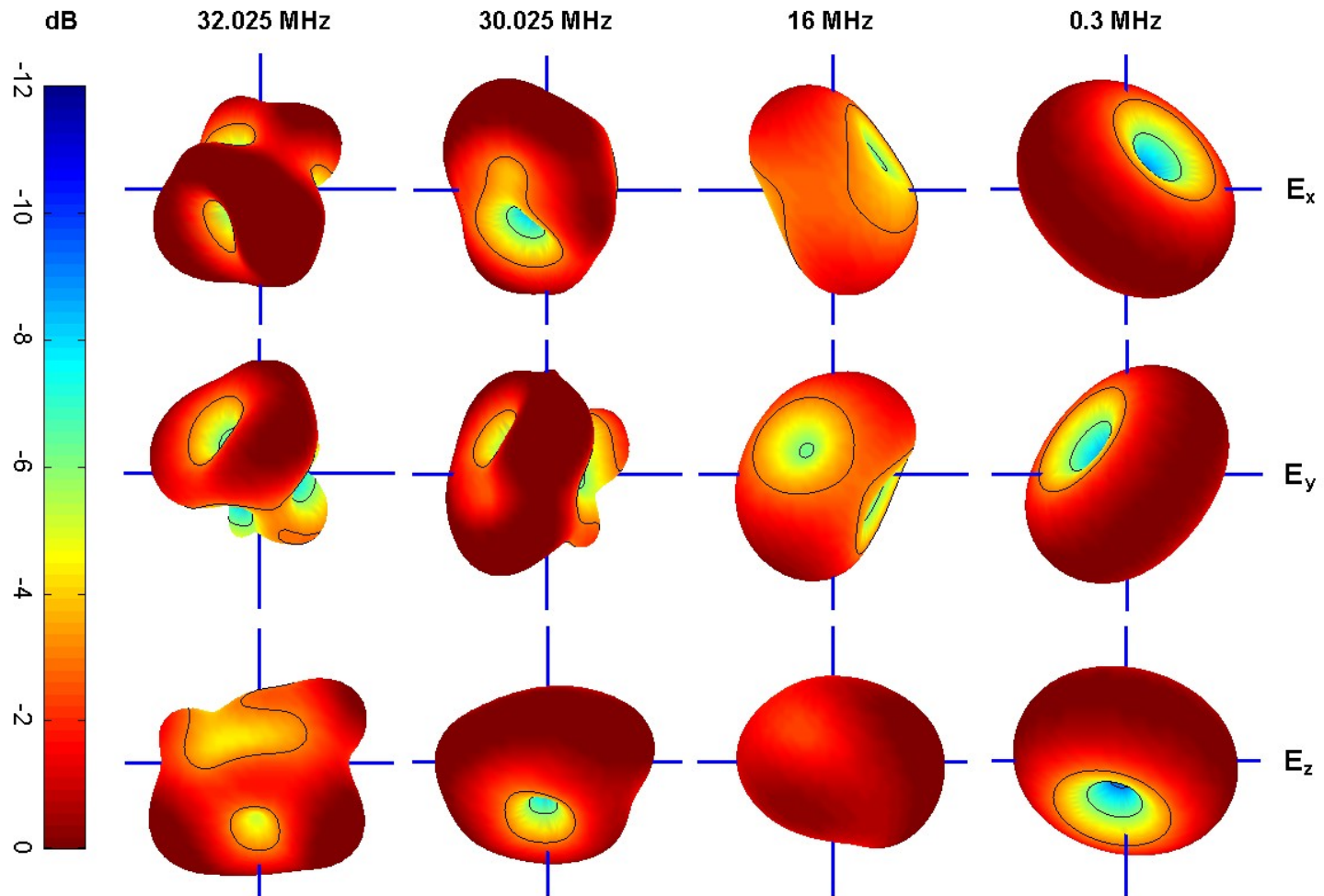


- Calculations were performed for open feeds and base capacitances of 90pF.
- A correction for the real antenna diameters has to be applied on the ASAP results. CONCEPT can deal with real antenna diameters.

# The STEREO/WAVES antenna calibration

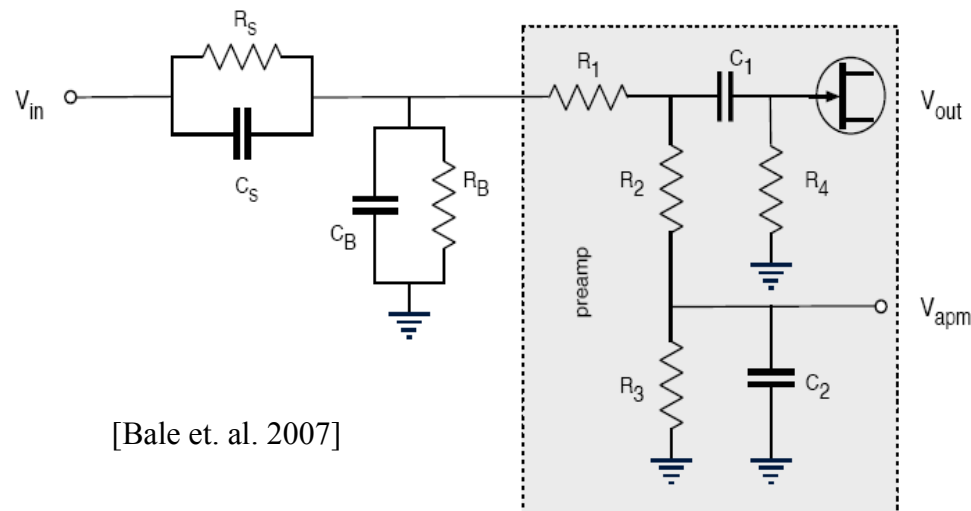
The effect of changing  
the model

Component	Antenna	$\delta h^\circ$ [m]	$\delta \zeta^\circ$ [deg]	$\delta \xi^\circ$ [deg]	Illustration
1. Battery	E <sub>x</sub>	0.01	0.1	0.2	
	E <sub>y</sub>	0.00	0.0	0.2	
	E <sub>z</sub>	0.00	0.1	0.0	
2. SECCHI (HI)	E <sub>x</sub>	0.00	0.0	0.1	
	E <sub>y</sub>	0.00	0.0	0.1	
	E <sub>z</sub>	0.00	0.1	0.0	
3. Beveling of hull edges	E <sub>x</sub>	0.01	0.0	0.2	
	E <sub>y</sub>	0.02	0.0	0.0	
	E <sub>z</sub>	0.01	0.0	0.3	
4. Change of antenna connections	E <sub>x</sub>	0.08	1.3	6.6	
	E <sub>y</sub>	0.02	0.2	0.1	
	E <sub>z</sub>	0.13	3.8	0.1	
5. Change of feed positions	E <sub>x</sub>	0.11	0.0	0.1	
	E <sub>y</sub>	0.12	0.1	0.1	
	E <sub>z</sub>	0.09	0.1	0.3	
6. Redesign of feed area	E <sub>x</sub>	0.00	0.0	0.0	
	E <sub>y</sub>	0.01	0.1	0.0	
	E <sub>z</sub>	0.00	0.0	0.0	
7. Redesign of HGA	E <sub>x</sub>	0.03	0.2	0.5	
	E <sub>y</sub>	0.03	0.1	0.4	
	E <sub>z</sub>	0.01	0.8	0.9	
8. Redesign of boom	E <sub>x</sub>	0.05	0.6	0.2	
	E <sub>y</sub>	0.03	0.6	0.3	
	E <sub>z</sub>	0.02	1.2	0.3	
9. Change of boom length by 1 m	E <sub>x</sub>	0.10	3.1	0.9	 boom length 5m → 6m
	E <sub>y</sub>	0.07	2.8	0.3	
	E <sub>z</sub>	0.16	3.0	1.1	



# The STEREO/WAVES antenna calibration

- Spacecraft antennas are usually coupled to the surrounding space plasma.
- The electromagnetic coupling can be modeled by a system of a resistance and a capacitance.
- In rarefied plasma the coupling can not take place without photoelectrons.
- The sheath thickness must not be larger than the antennas.



[Bale et. al. 2007]

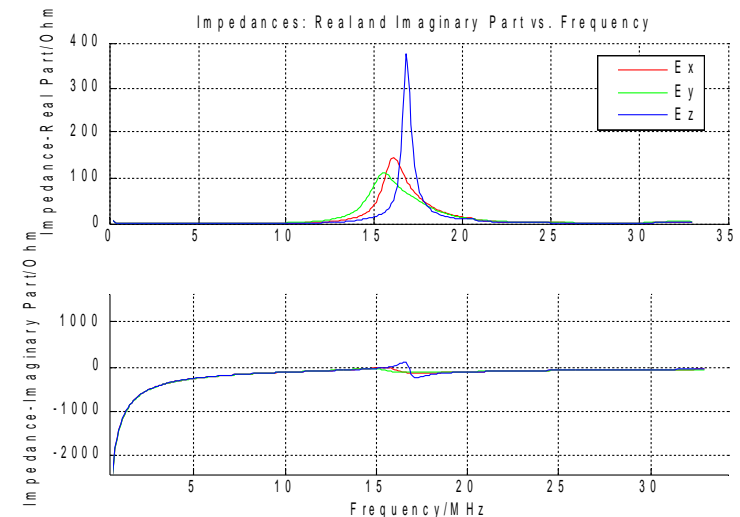
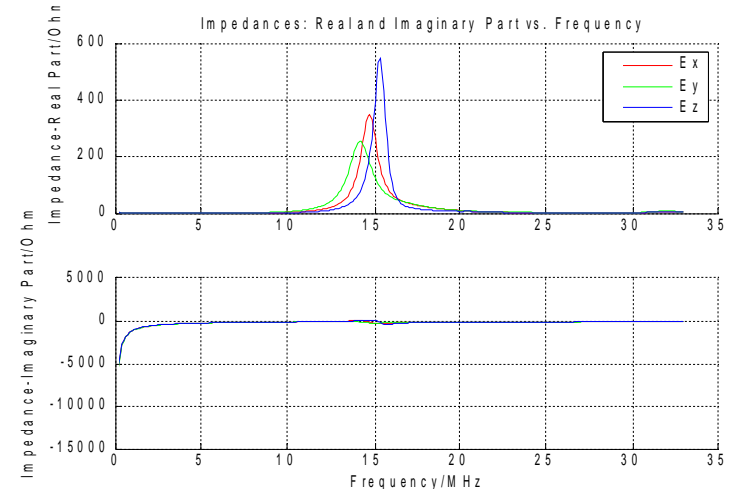
- STEREO operates in solar wind conditions at 1AU.
- The photo-electron production rate is higher than the thermal electron impact rate.  
--> positive charge.
- $i_{ph} \dots 10^{-4} \text{Am}^{-2}$  [Fahrleson 1967]
- $A_{rel} \dots 0.5$
- $l = 6\text{m}$
- $d = 1\text{in}$  (0.0254m) on average
- Mean energy of photo-electrons = 1.5eV [Grard 1973]
- Mean energy of thermal electrons = 10eV

$$I_{ph} = A_{rel} i_{ph} l d \sim 7.6 \cdot 10^{-6} \text{A}$$

$$I_e = -e n_e d \pi \sqrt{\frac{\kappa T_e}{2\pi m_e}} \sim -2 \cdot 10^{-7} \text{A}$$

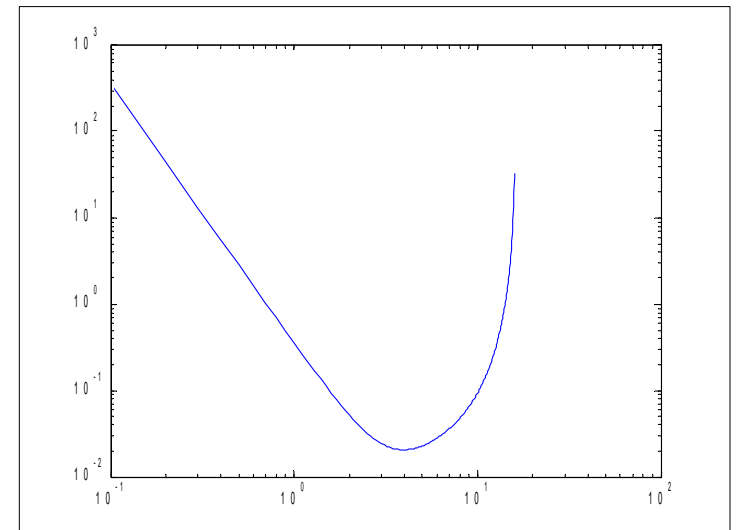
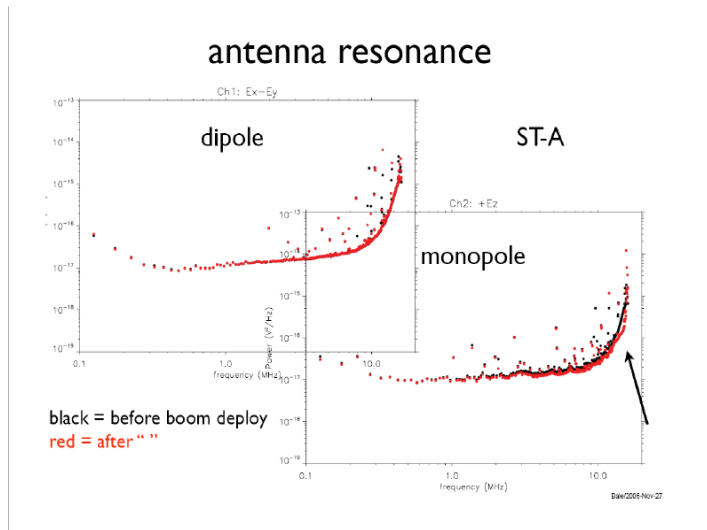
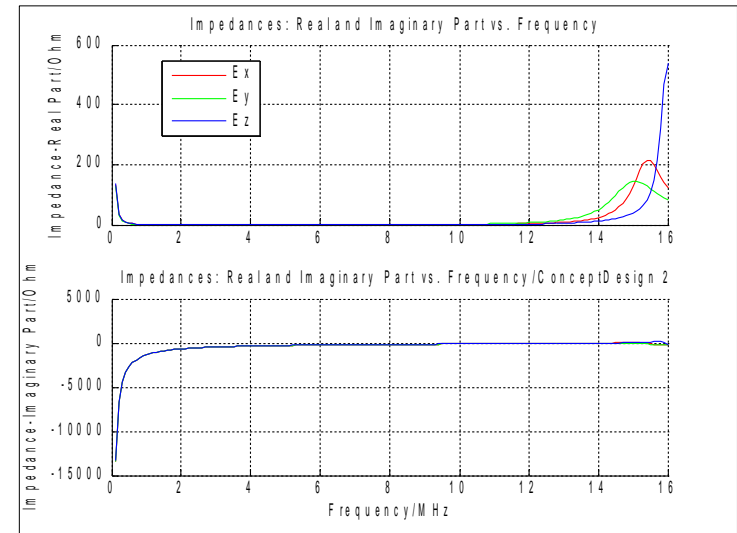
- >  $V = 5.5\text{V}$
- $\bar{n}_e = 10^6 \text{m}^{-3}$
- >  $n_{ph}(0) = 2 \times 10^8 \text{m}^{-3}$
- > 2.5% of the photo-electrons reach the plasma.
- >  $\lambda_{sh} = 0.6\text{m}$  or  $0.4\text{m}$ , depending on the method.

- Using the appropriate equations, one finds:
  - $R_s = 0.2 \text{ M}\Omega$
  - $C_s = 87 \text{ pF}$
- Via these parameters the sheath can be included into the numerical antenna calibration (wire-grid).
- Computation of the impedances show that the inclusion of the sheath has an effect.





- Comparing the location of the second resonance of antenna  $E_z$  with measured data shows that the inclusion of the sheath capacitance has a corrective effect.
- Further and more accurate data is needed to verify the model.



Thank You for Your attention !