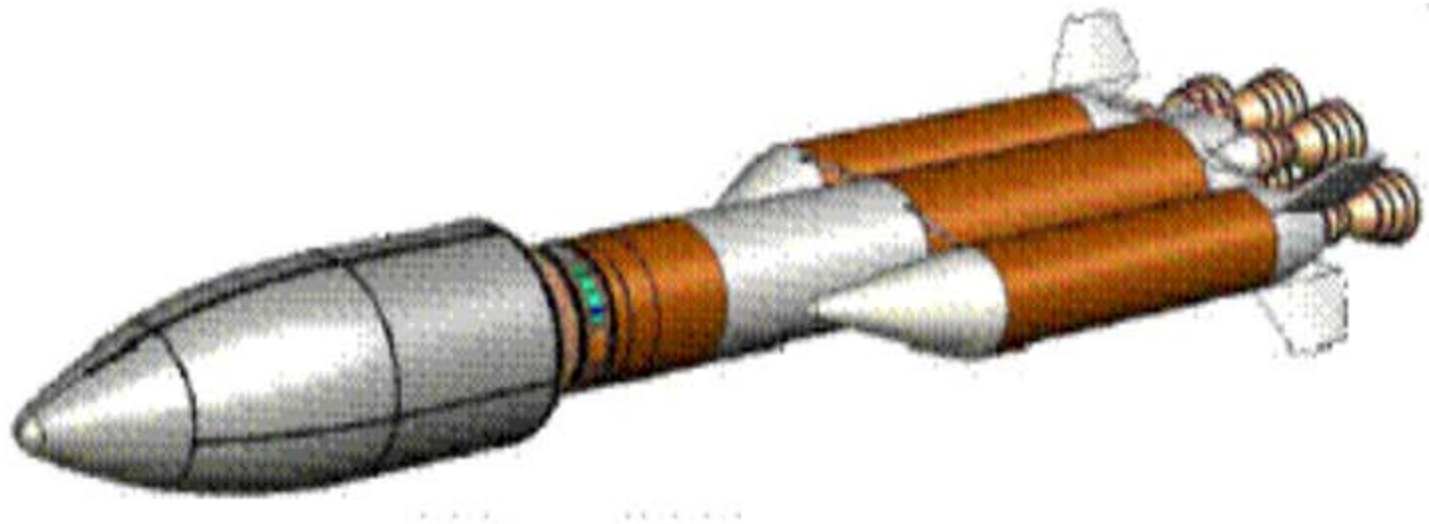
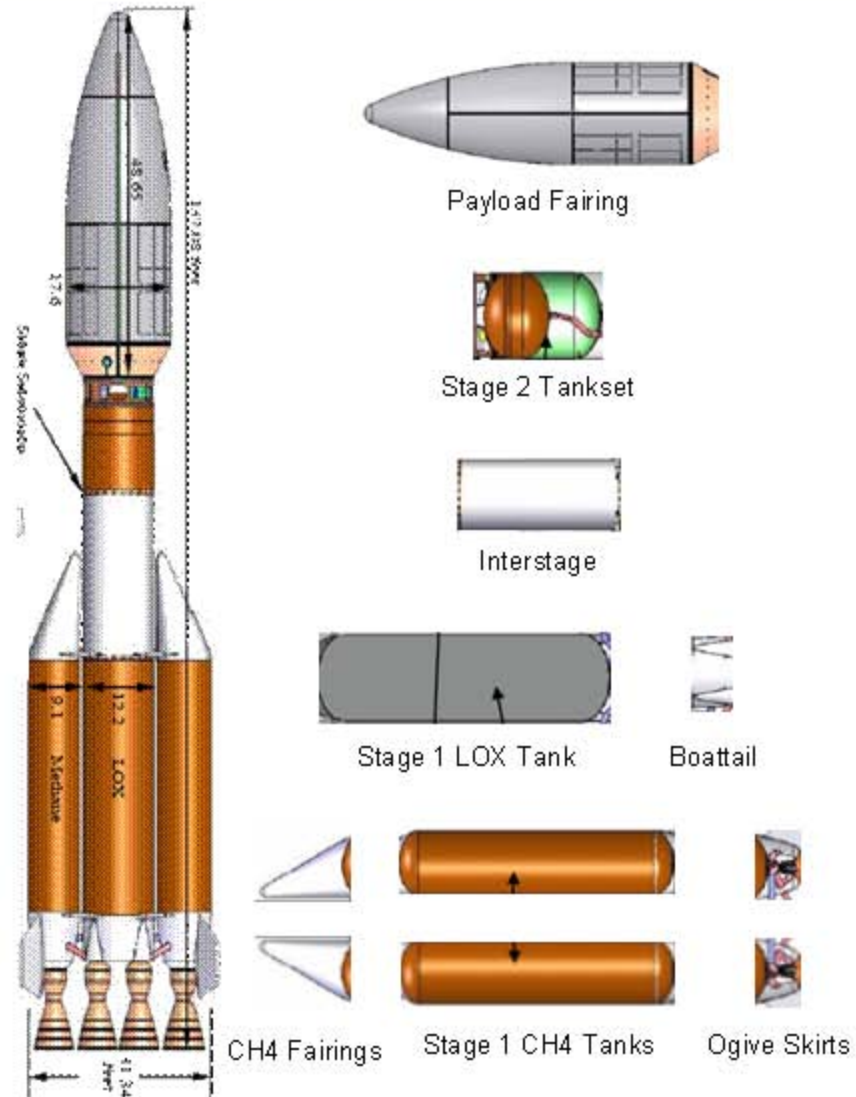


AL PLP Heating Analysis Results



AirLaunch PLP

AL PLP Heating Analysis Results



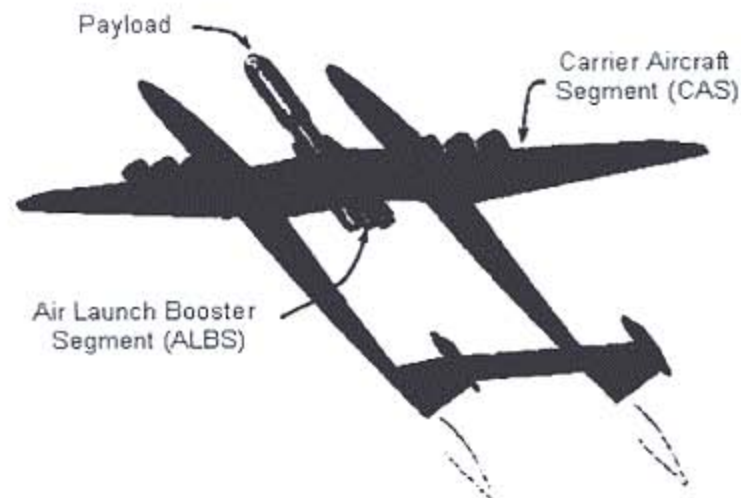
Launch Description

Objective:

- Develop and flight test an innovative new capability to transport medium class (12,000 Lbs minimum) payloads to low earth orbit

Requirements:

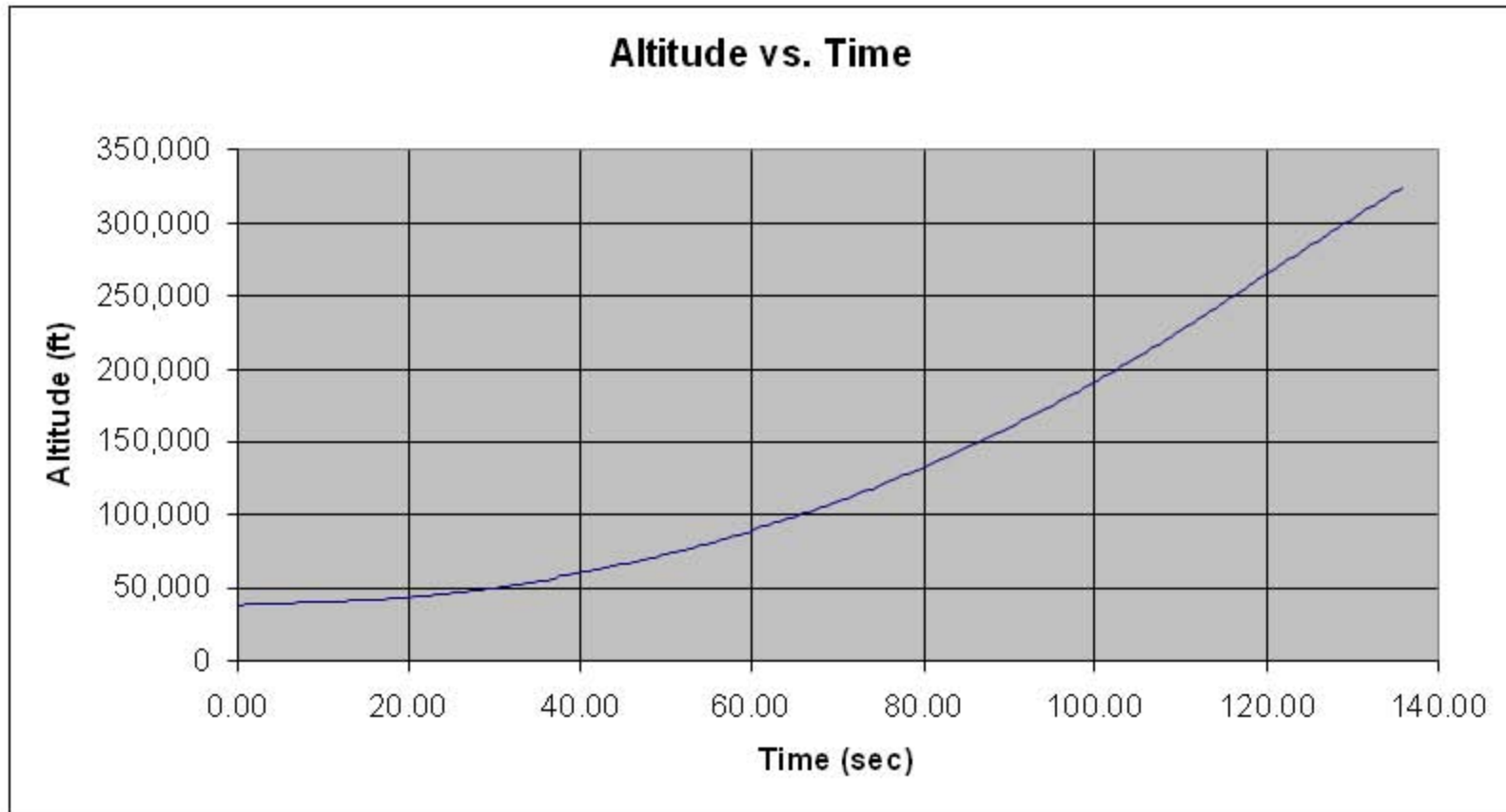
- 12,000 Payload
- 550,000 LB GLOW (ALBS and ASE)
- Orbit Insertion 250 NM / 60 degree inclination
- Mated Duration: 3 Week with multiple launch attempts without demating
- Launch 14 Days After Payload Delivery
- Mate and launch within 4 days of Carrier Aircraft availability
- Abort and Recover with ALBS with next launch attempt within 24 hours



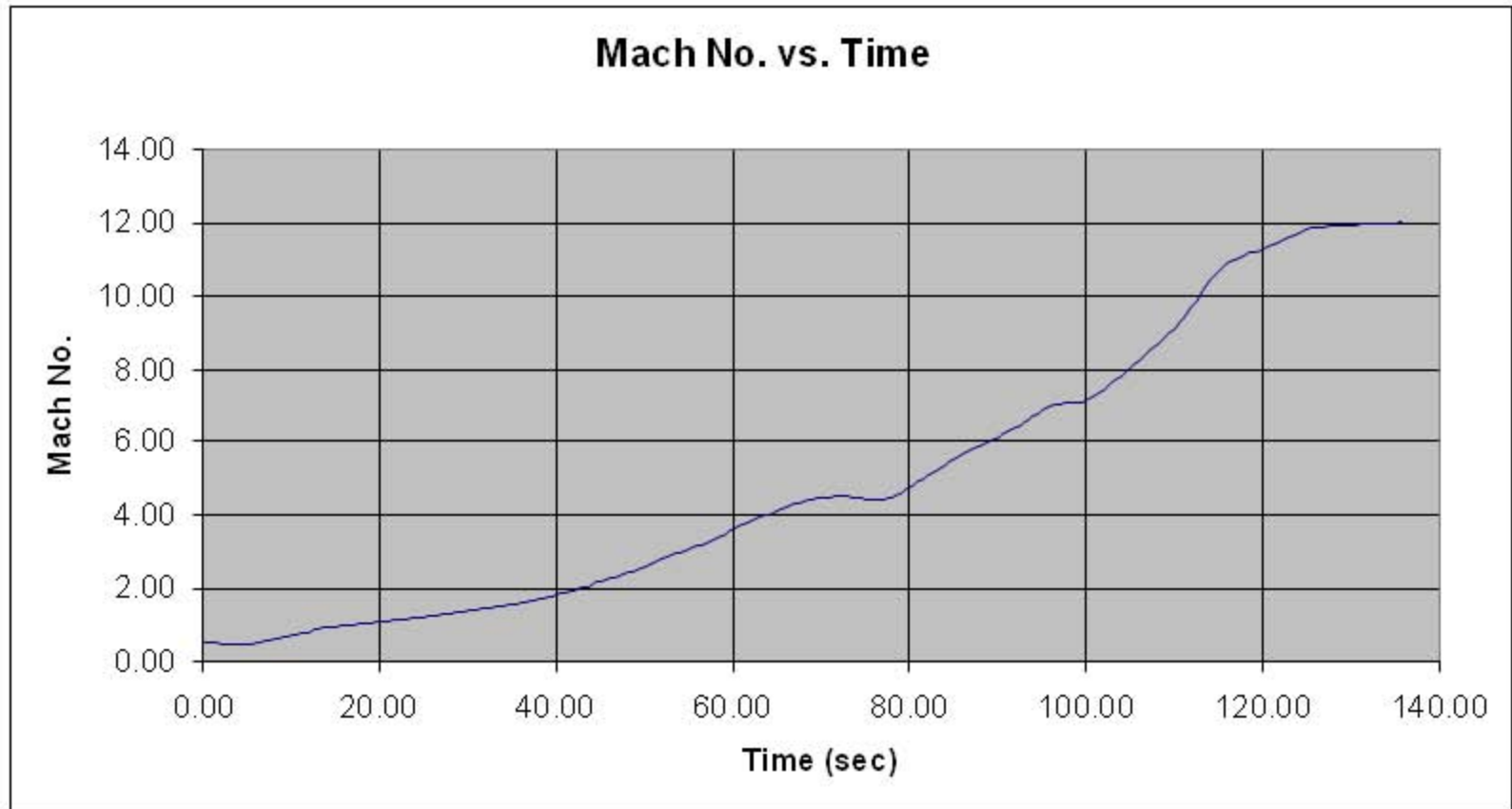
Example Release Conditions

- 450K lb booster
- 33K ft Maneuver Start Altitude
- 43 Degrees Gamma (Nose Up)
- Mach 0.55 to 0.65
- 42,000 Ft

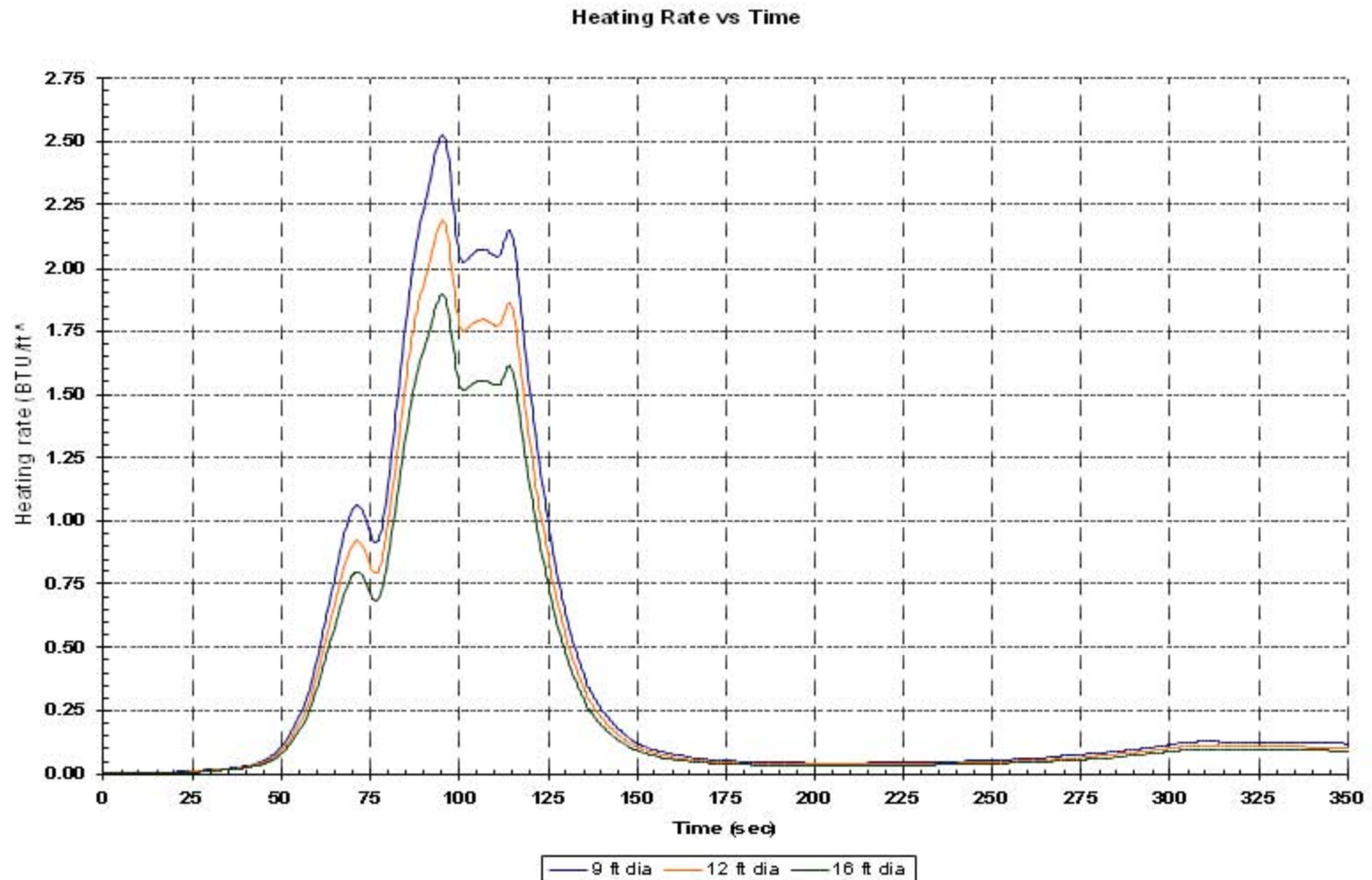
AL PLP Heating Analysis Results



AL PLP Heating Analysis Results



AL PLP Heating Analysis Results



AL PLP Heating Analysis Results

- Cross Section:
 - Outer Skin: 0.078" Graphite Epoxy
 - Core: 1" Vacuum Purged, Perforated Aramid Honeycomb
 - Inner Skin: 0.078" Graphite Epoxy
- No Exterior Coating (Emissivity: 0.80 @ -460 degF to 0.90 @ 4000 degF)
- Tank Interior Boundary Conditions:
 - LOX: -234 degF @ 200 psia to -245 degF at 140 psia (70% P_{i_LOX})
 - CH₄: -178 degF @ 200 psia to -193 degF at 140 psia (70% P_{i_CH4})
- Pre-condition w/ Exterior Convection Prior to Launch ($h = 20 \text{ btu/hr-ft}^2\text{-F}$, 1962 Standard Day Atmospheric Model)
- Post-Launch:
 - Applied Exterior Aero-Thermal B.C.
 - No Additional Exterior Convection
 - Radiation to T_{sky} as Function of Altitude
 - Radiation to T_{earth} as Function of Altitude
 - Solar Load as Function of Altitude

AL PLP Heating Analysis Results

Mission Time: 0 to 135 sec.

- 9 Ft. Dia. CH₄ Tank:
 - T_{max} Outer Skin: 555 F / Core: 137 F / Inner Skin: -114 F
 - T_{min} Outer Skin: -51 F / Core: -90 F / Inner Skin: -144 F
- 12 Ft. Dia. CH₄ Tank:
 - T_{max} Outer Skin: 507 F / Core: 110 F / Inner Skin: -117 F
 - T_{min} Outer Skin: -51 F / Core: -90 F / Inner Skin: -144 F
- 9 Ft. Dia. LOX Tank:
 - T_{max} Outer Skin: 554 F / Core: 127 F / Inner Skin: -155 F
 - T_{min} Outer Skin: -53 F / Core: -106 F / Inner Skin: -188 F
- 12 Ft. Dia. LOX Tank:
 - T_{max} Outer Skin: 505 F / Core: 100 F / Inner Skin: -159 F
 - T_{min} Outer Skin: -53 F / Core: -107 F / Inner Skin: -188 F
- 16 Ft. Dia. Fairing (Adiabatic Interior Surface):
 - T_{max} Outer Skin: 468 F / Core: 125 F / Inner Skin: -10 F
 - T_{min} Outer Skin: -46 F / Core: -45 F / Inner Skin: -44 F

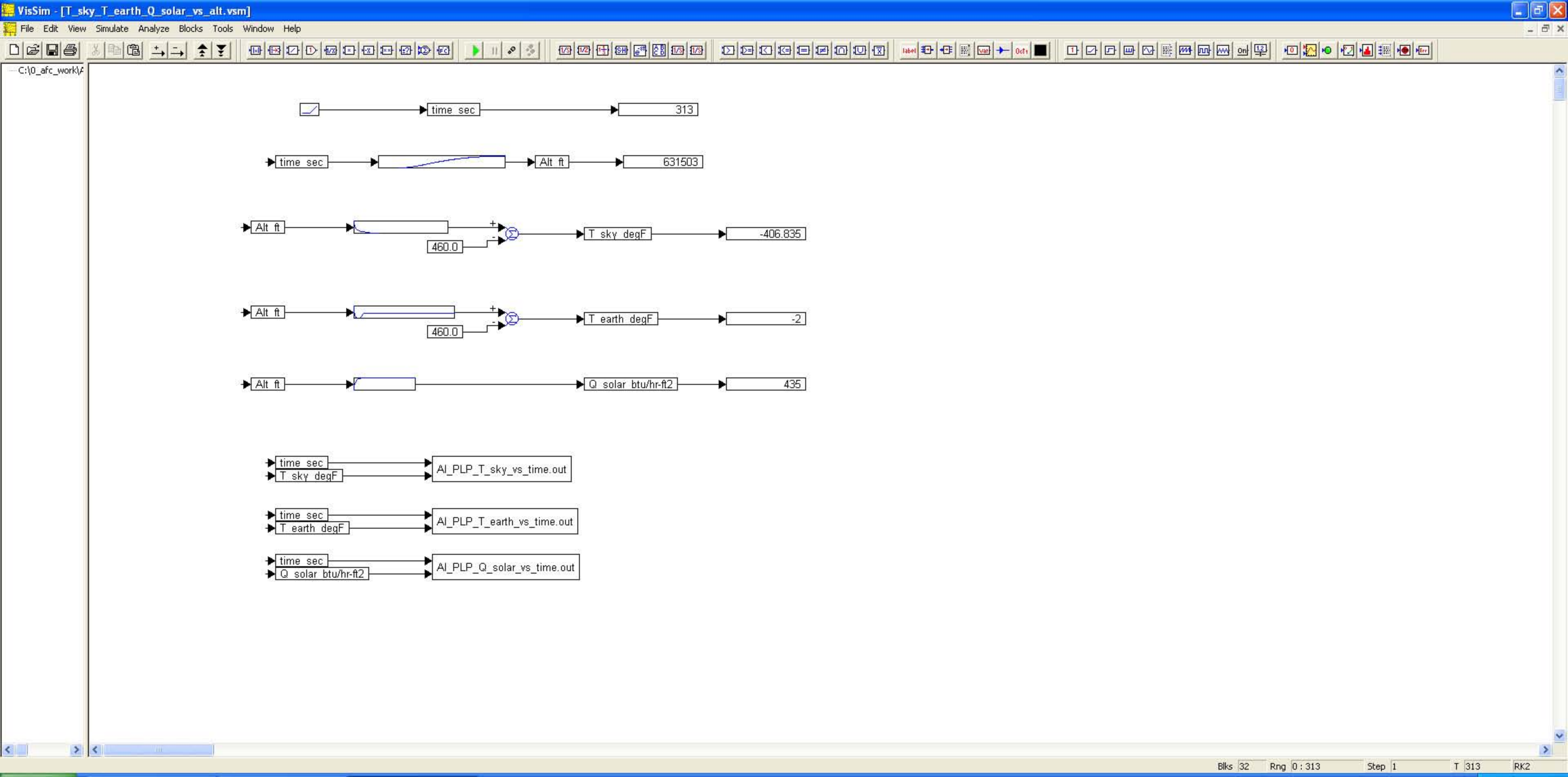
AL PLP Heating Analysis Results

CONCLUSIONS:

- Outer Skin Exceeds Desired 250 degF Maximum Subsequent to $t = 85$ sec. (i.e. $T_{\max} = 555$ degF)
- Inner Skin Does Not Exceed 250 degF Maximum (i.e. $T_{\max} = 137$ degF)

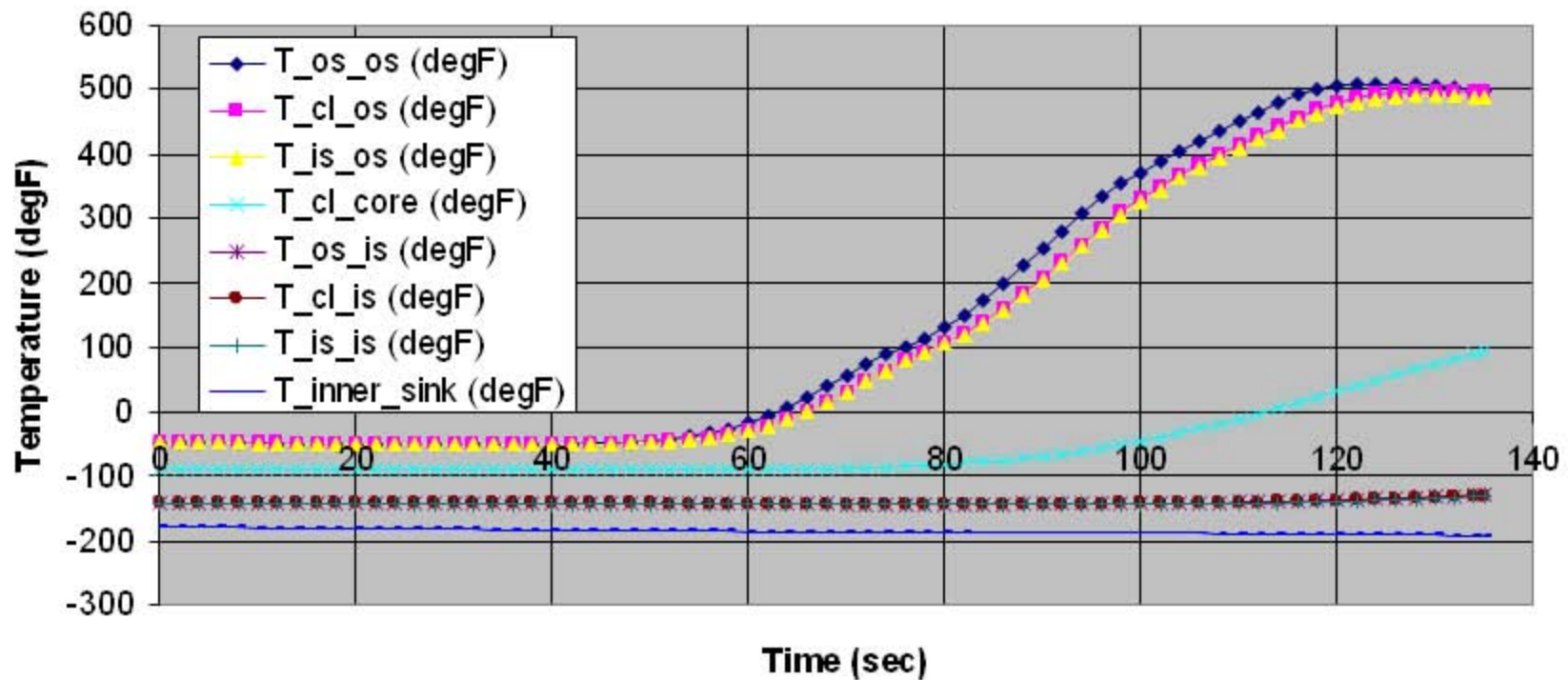
SUGGESTION:

- Employ an Asymmetric Cross Section:
 - Outer Skin + Core + Inner Skin Sized to Provide Structural Margins During Handling and Transport Loads (i.e. $-\infty < t < 85$ sec.)
 - Inner Skin Alone Sized to Provide Structural Margins Subsequent to Launch (i.e. $t > 85$ sec.)
- Currently Looking at Various Thermal Barrier Coatings



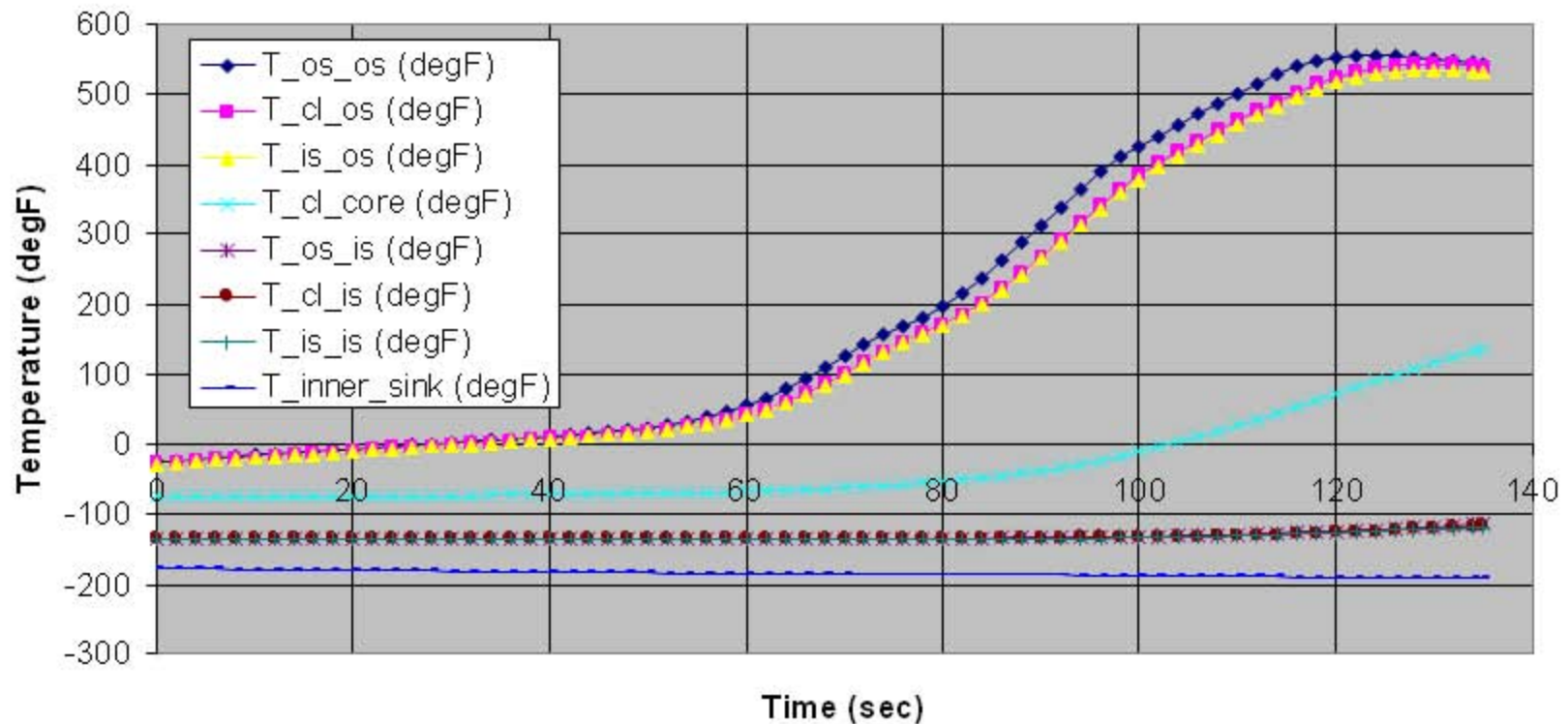
AL PLP Heating Analysis Results

Figure 1: 9 ft. Dia. CH₄ Tank w/o Solar Load, w/o Earth Radiative Coupling



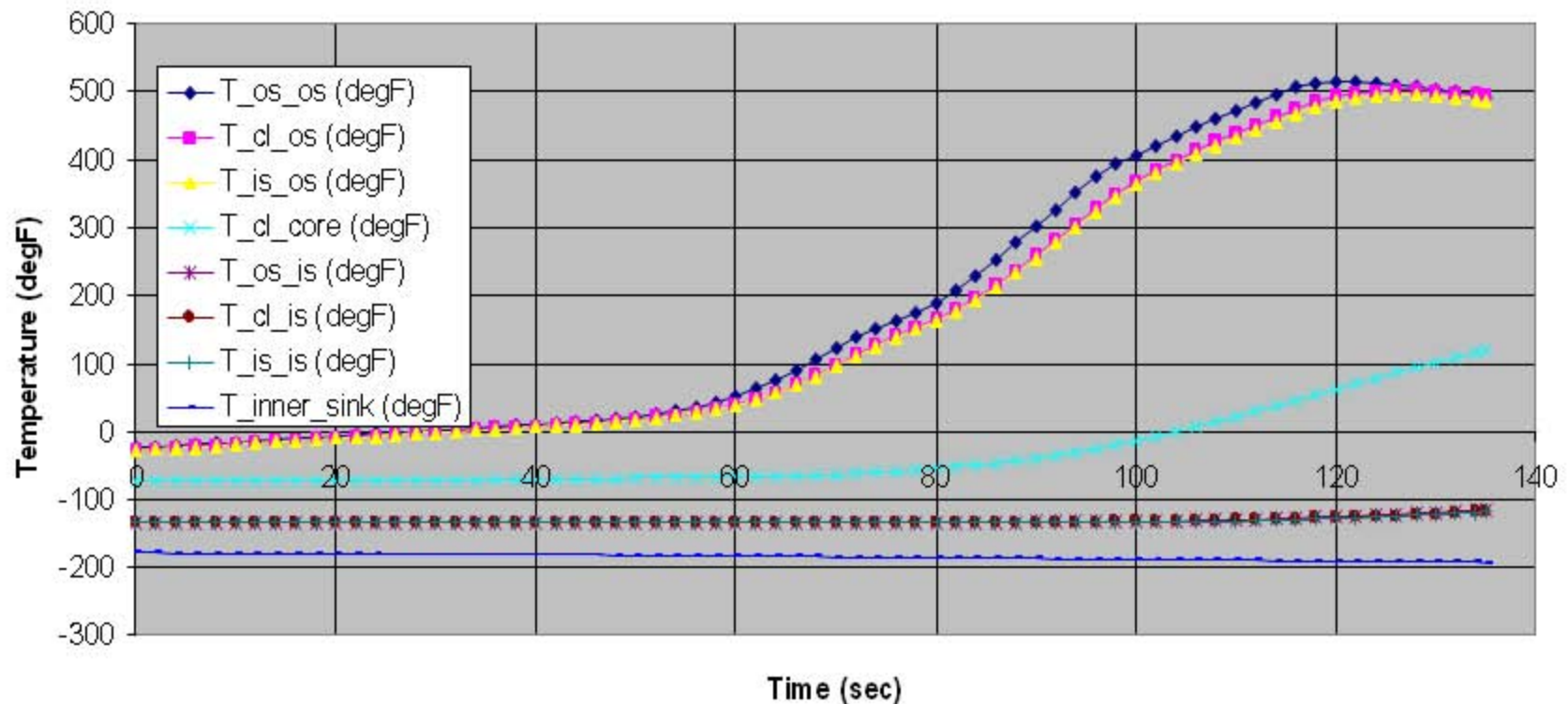
AL PLP Heating Analysis Results

Figure 2: 9 ft. Dia. CH₄ Tank w/Solar Load, w/o Earth Radiative Coupling



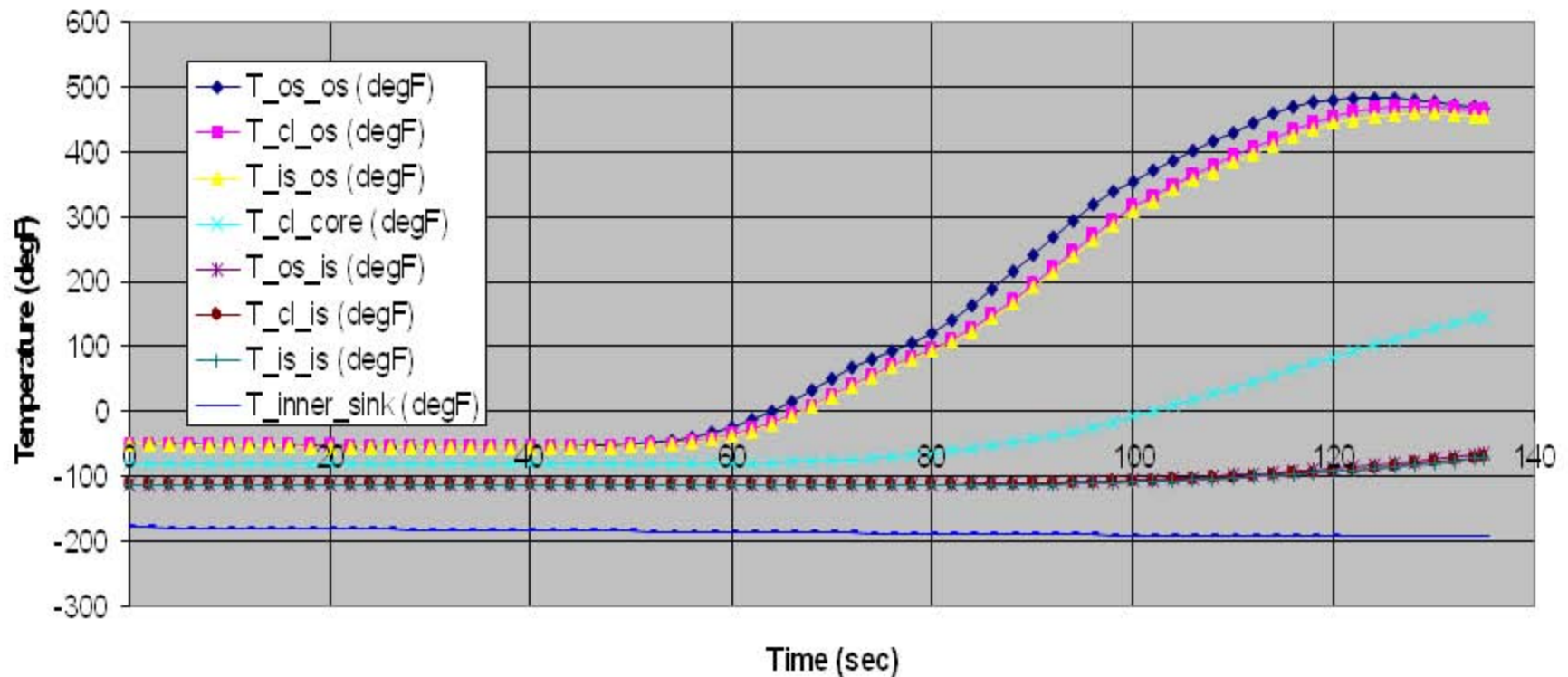
AL PLP Heating Analysis Results

Figure 3: 9 ft. Dia. CH₄ Tank w/Solar Load, w/ Earth Radiative Coupling



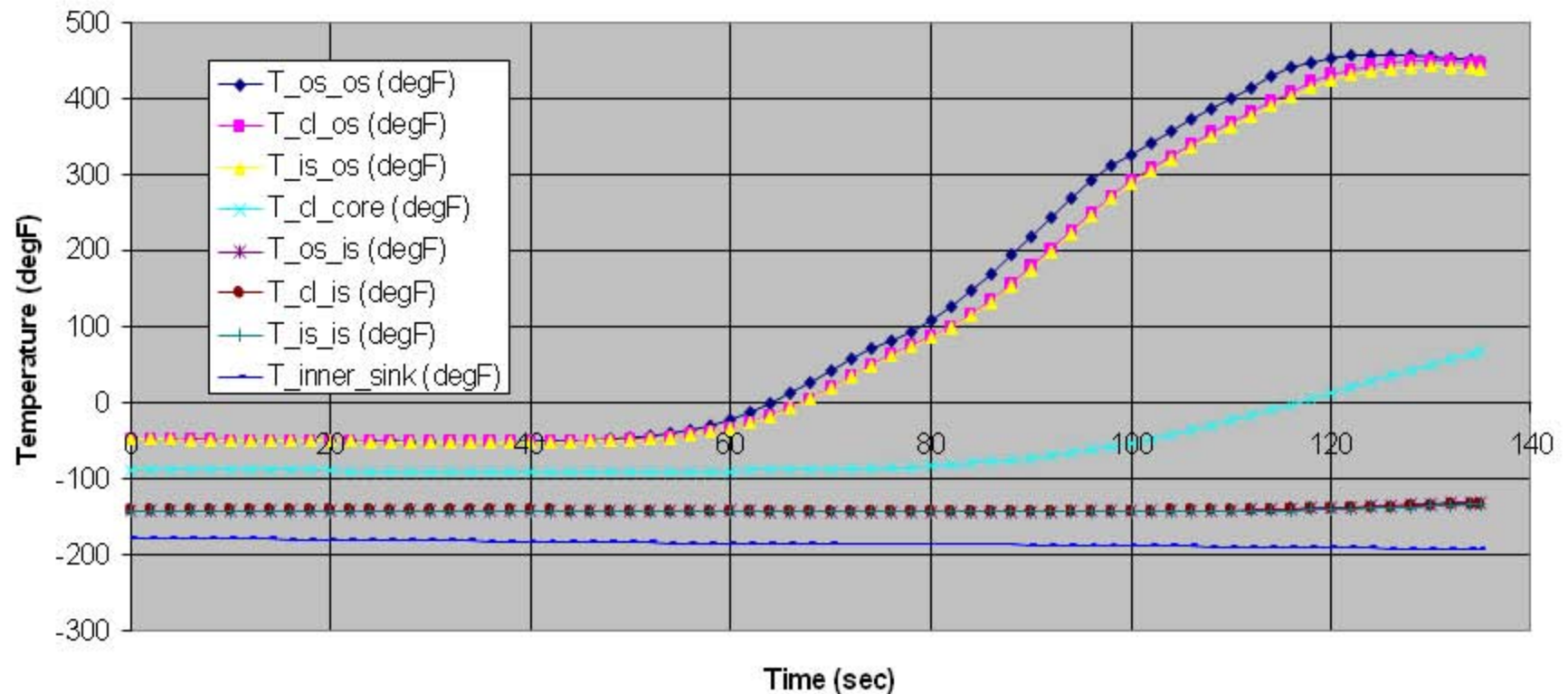
AL PLP Heating Analysis Results

**Figure 4: 9 ft. Dia. CH₄ Tank w/o Solar Load, w/o Earth Radiative Coupling
(Core Material Conductivity Increased by approximate factor of 10)**



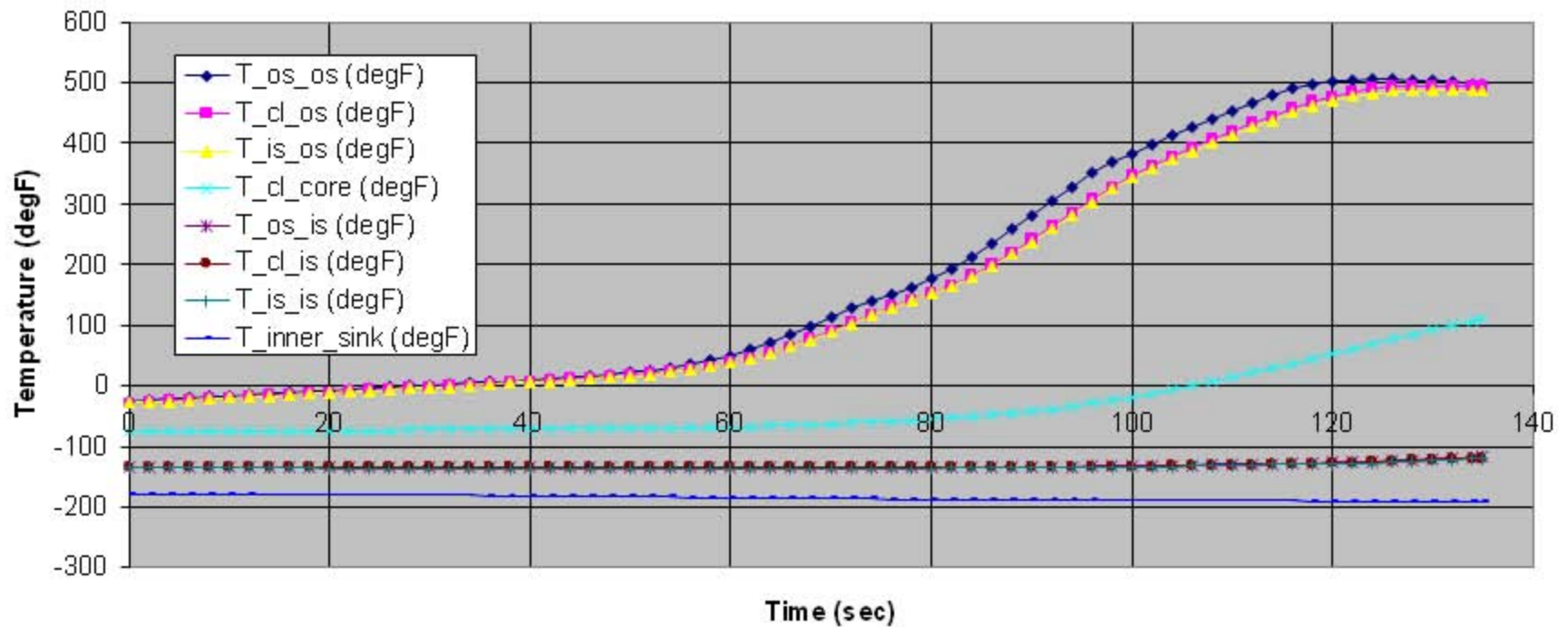
AL PLP Heating Analysis Results

Figure 5: 12 ft. Dia. CH₄ Tank w/o Solar Load, w/o Earth Radiative Coupling



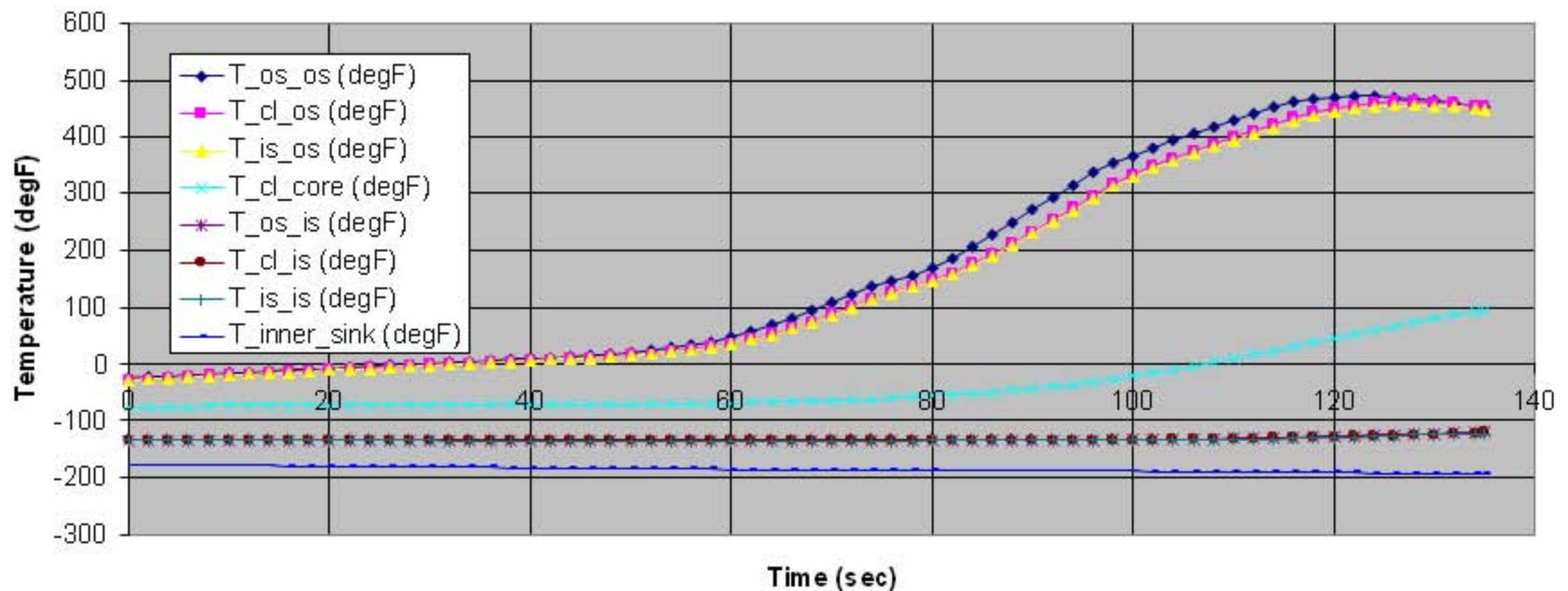
AL PLP Heating Analysis Results

Figure 6: 12 ft. Dia. CH₄ Tank w/ Solar Load, w/o Earth Radiative Coupling



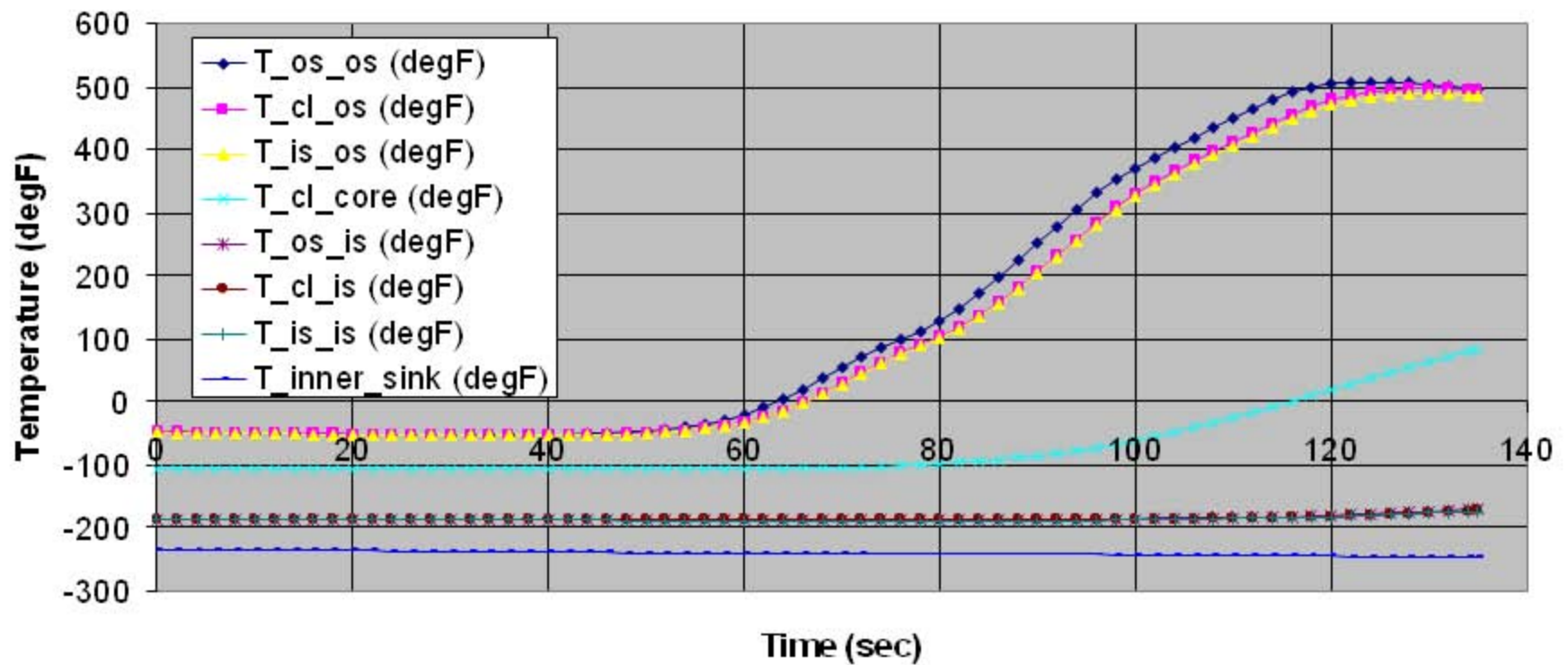
AL PLP Heating Analysis Results

Figure 7: 12 ft. Dia CH₄ Tank w/ Solar Load, w/ Earth Radiative Coupling



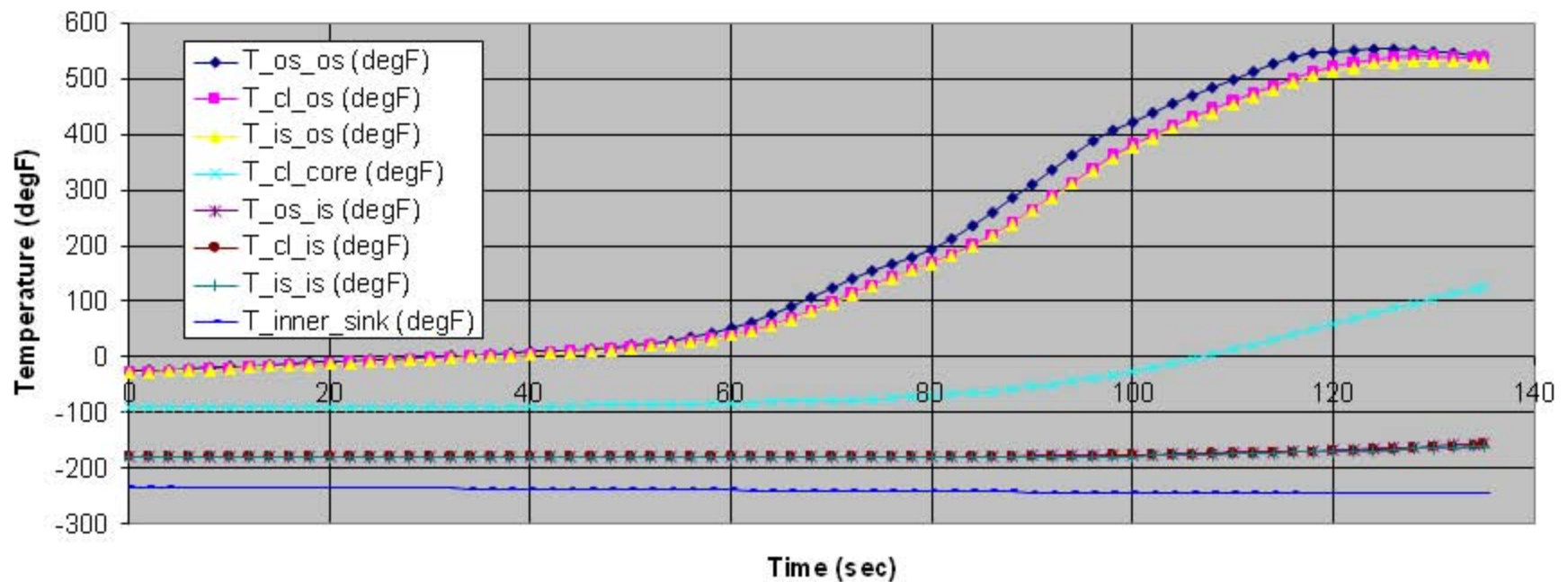
AL PLP Heating Analysis Results

Figure 8: 9 ft. Dia. LOX Tank w/o Solar Load, w/o Earth Radiative Coupling



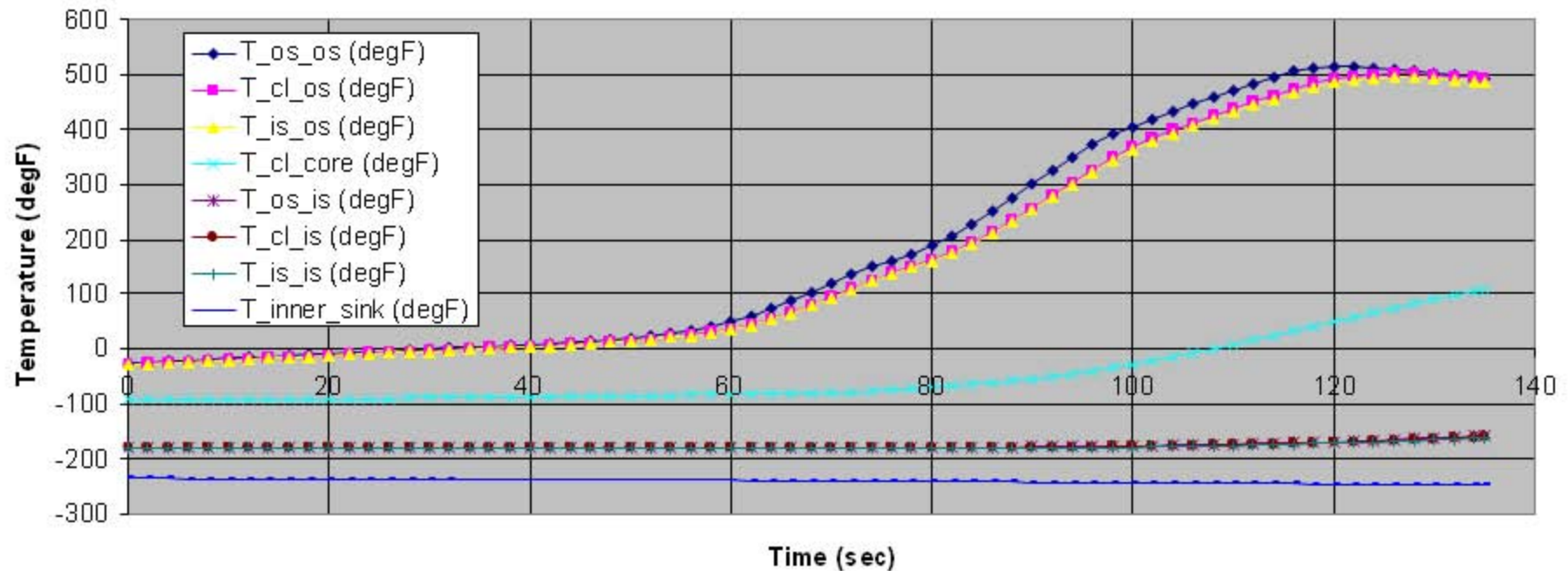
AL PLP Heating Analysis Results

Figure 9: 9 ft. Dia. LOX Tank w/ Solar Load, w/o Earth Radiative Coupling



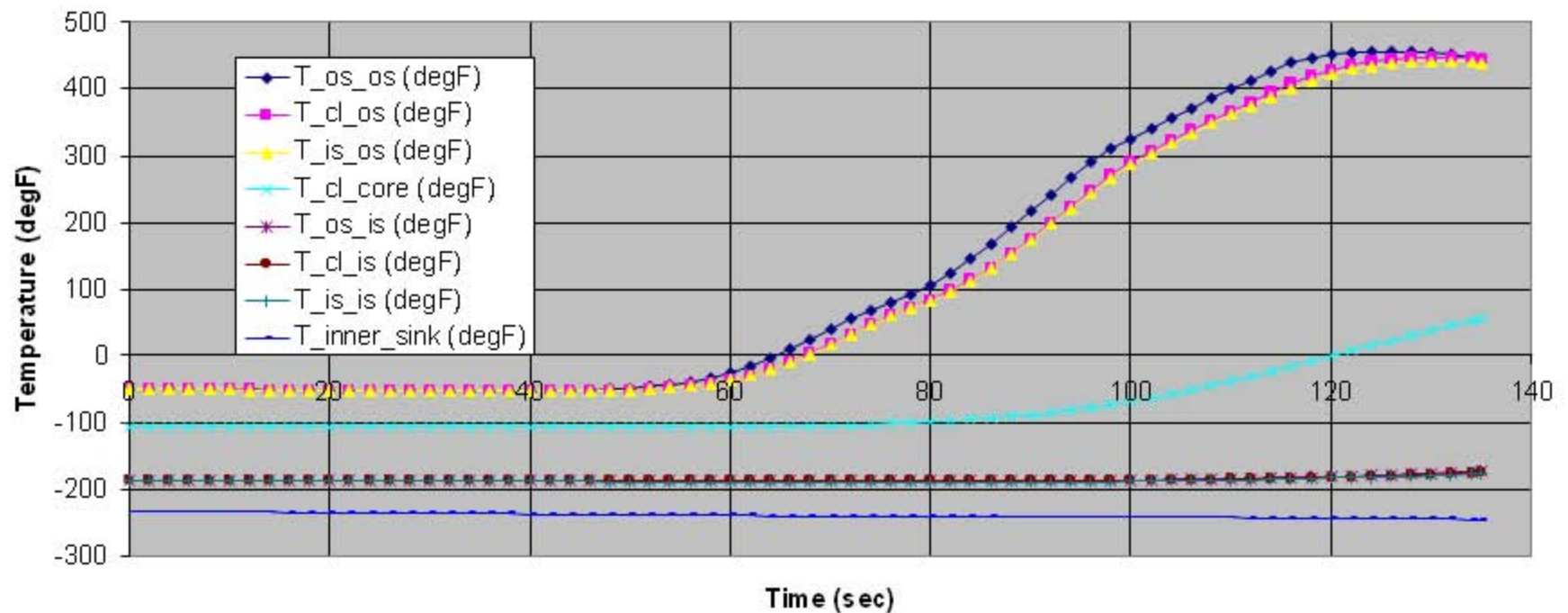
AL PLP Heating Analysis Results

Figure 10: 9 ft. Dia. LOX Tank w/ Solar Load, w/ Earth Radiative Coupling



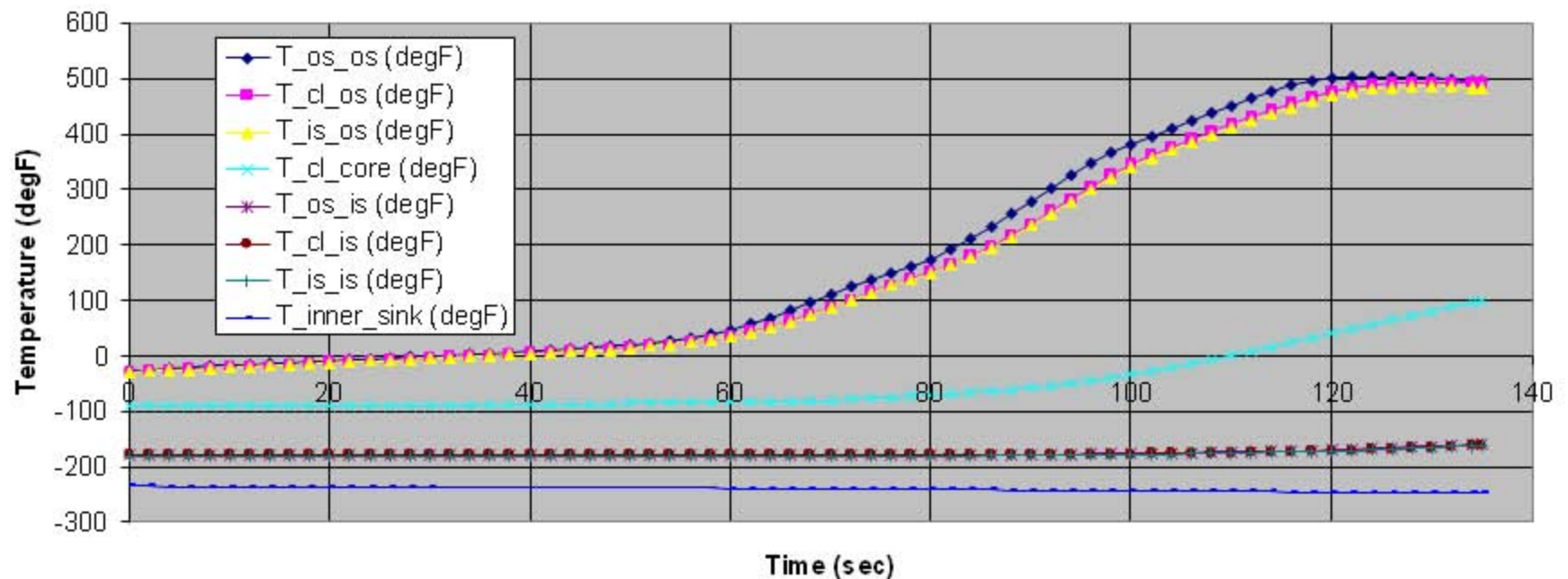
AL PLP Heating Analysis Results

Figure 11: 12 ft. Dia. LOX Tank w/o Solar Load, w/o Earth Radiative Coupling



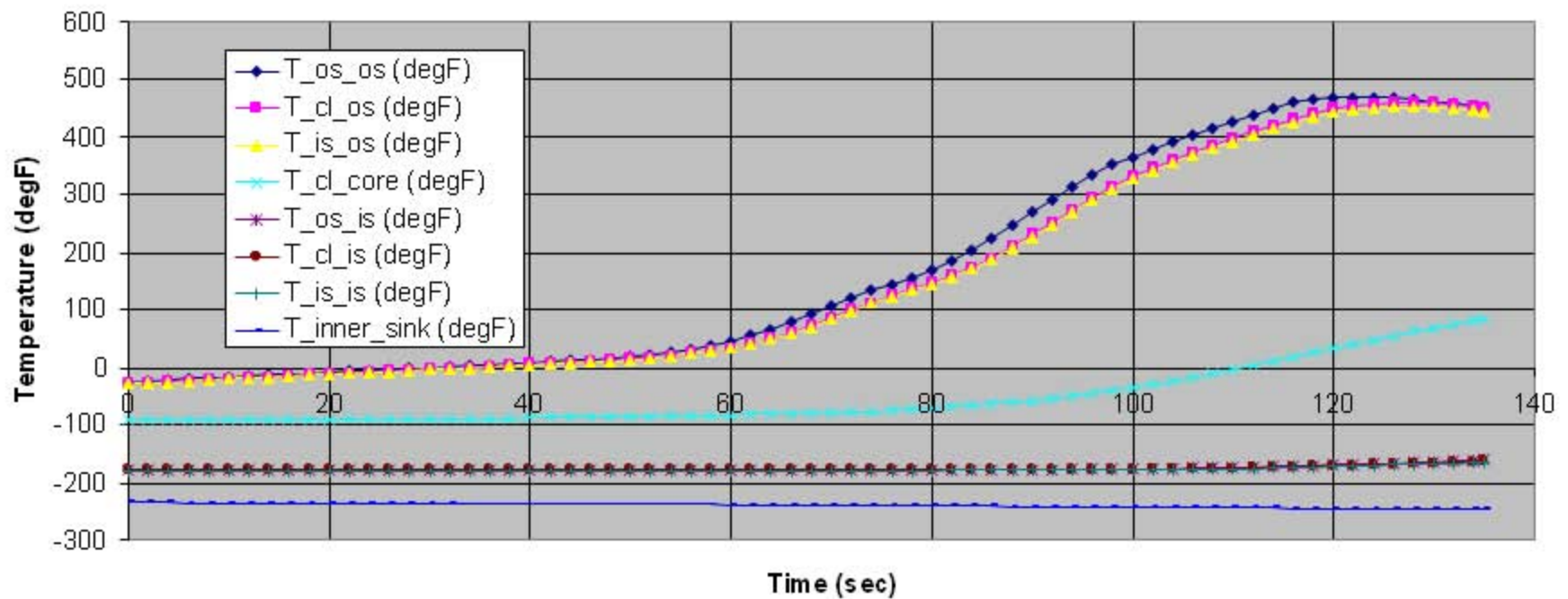
AL PLP Heating Analysis Results

Figure 12: 12 ft. Dia. LOX Tank w/ Solar Load, w/o Earth Radiative Coupling



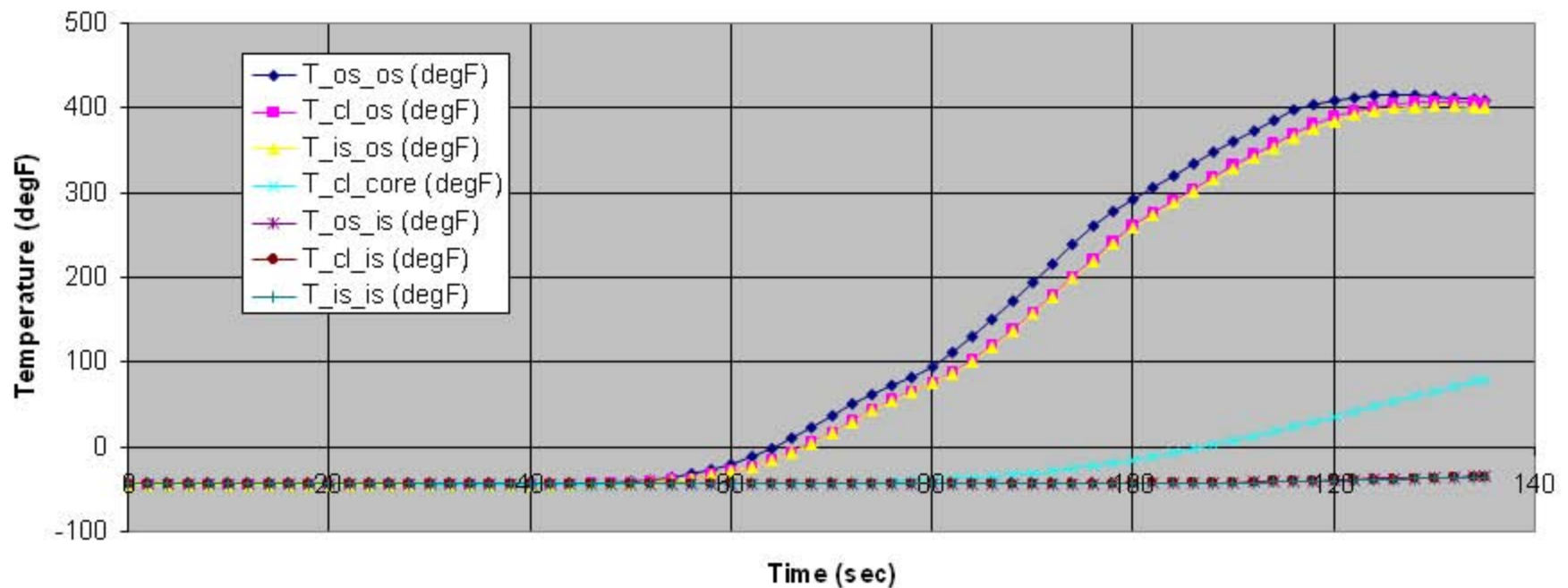
AL PLP Heating Analysis Results

Figure 13: 12 ft. Dia. LOX Tank w/ Solar Load, w/ Earth Radiative Coupling



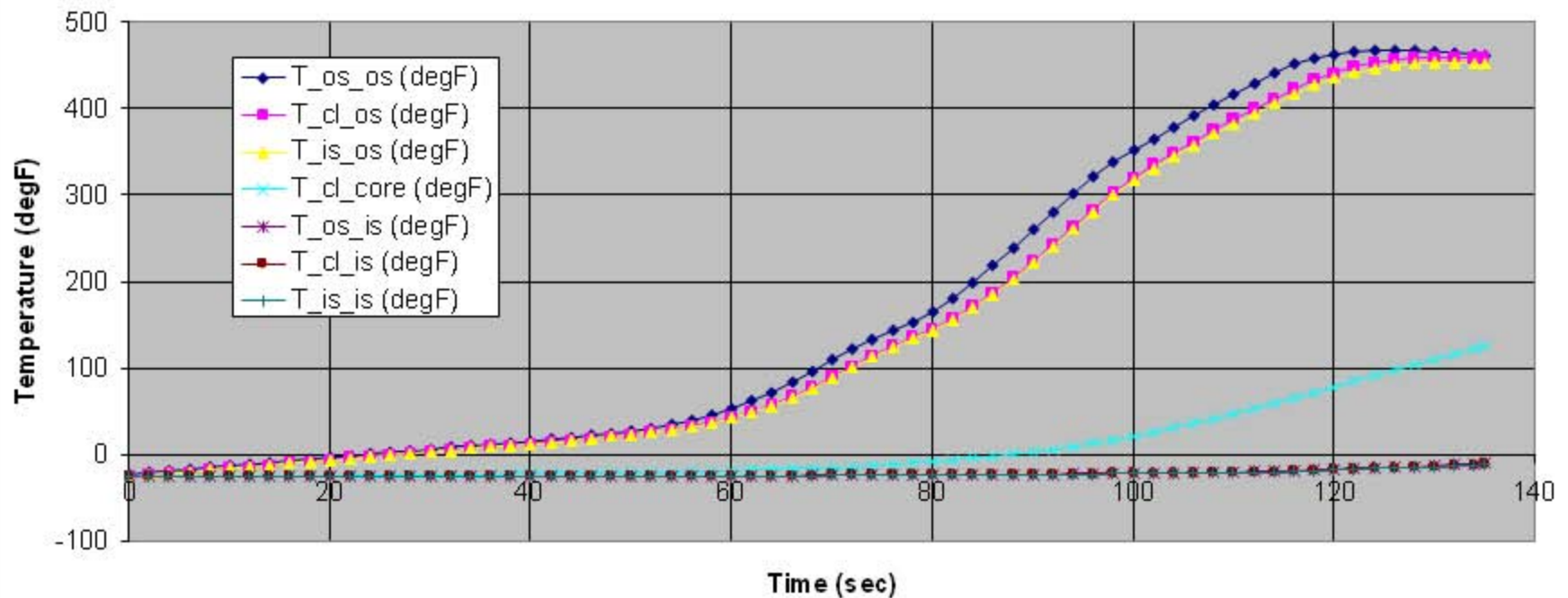
AL PLP Heating Analysis Results

Figure 14: 16 ft. Dia. Fairing w/o Solar Load, w/o Earth Radiative Coupling



AL PLP Heating Analysis Results

Figure 15: 16 ft. Dia. Fairing w/ Solar Load, w/o Earth Radiative Coupling



AL PLP Heating Analysis Results

Figure 16: 16 ft. Dia. Fairing w/ Solar Load, w/ Earth Radiative Coupling

