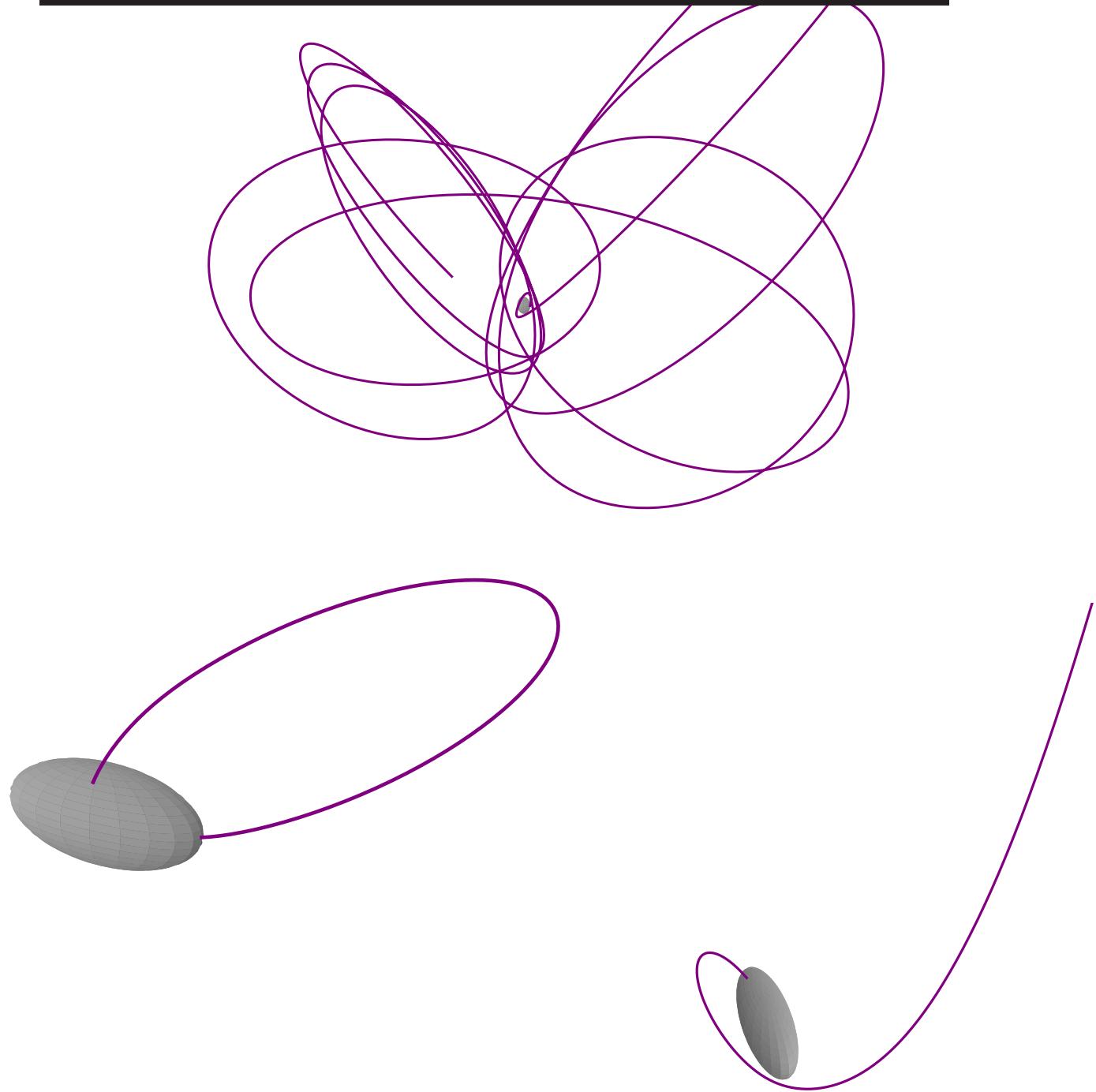


Orbital motion of regolith around asteroids

MSc Thesis Report

Abhishek Agrawal



ORBITAL MOTION OF REGOLITH AROUND ASTEROIDS

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by

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"If you wish to make an apple pie from scratch, you must first invent the universe."

Carl Sagan

ACKNOWLEDGMENTS

ABSTRACT

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LIST OF SYMBOLS

ROMAN

Symbol	Units	Description
a, b, c	m	Semi-axes of reference ellipsoid (ellipsoidal harmonics gravity potential model)
\mathcal{B}	-	Collection of all discrete mass distributions
C	-	Spherical harmonics expansion coefficient
E	-	Lamé function of first kind
E	-	Edge Dyad in the polyhedron potential model
F	-	Lamé function of second kind
F	-	Facet dyad in the polyhedron potential model
G	$m^3 g^{-1} s^{-2}$	Universal Gravitational constant
L_e	-	Per-edge factor in polyhedron model
\hat{n}	-	Unit normal vector
\hat{n}_e^f	-	Unit normal vector to edge e of facet f of a polyhedron
P	-	Associated Legendre functions
\mathbf{P}	-	Field point in polyhedron potential model
r	m	Distance
\vec{r}	m	Position vector
\vec{r}_i^f	m	Vector from field point to vertices of a facet in polyhedron model
\vec{r}_i^e	m	Vector from field point to vertices of an edge in polyhedron model
\vec{r}_e, \vec{r}_f	m	Vector from any point on edge e or f , respectively, to the field point in polyhedron model
R_D, R_F	-	Carlson elliptic integral functions
s	m	Conic equation parameter
S	-	Spherical harmonics expansion coefficient
t	s	Time
u	m	Parameter in equation for family of confocal quadrics to an ellipsoid (CDE potential model)

u_1, u_2, u_3	m	Real valued and unequal roots to the equation for family of confocal quadrics to an ellipsoid (CDE potential model)
U	m^2/s^2	Gravitational potential
U_1, U_2, U_3, U_4	m^2/s^2	CDE gravitational potential split in 4 parts
U_*	m/s^2	Gravitational acceleration component where $(* = x, y, z)$
v	m	Substitution parameter to get standard Carlson elliptic integral
v	m/s	Velocity
\vec{v}	m/s	Velocity vector
V	m^3	Volume
x	m	Cartesian coordinate x
y	m	Cartesian coordinate y
z	m	Cartesian coordinate z

GREEK

Symbol	Units	Description
α	-	Ellipsoidal harmonics expansion coefficient
α, β, γ	m	Semi-major axes of an ellipsoid (CDE potential model)
δ	deg	Latitude (spherical harmonics gravitational potential model)
Δ	-	Function in CDE potential model
λ	deg	Longitude (spherical harmonics gravitational potential model)
$\lambda_1, \lambda_2, \lambda_3$	m	Ellipsoidal coordinates
λ_1^{ref}	m	Largest semi-major axis of reference ellipsoid (ellipsoidal harmonics gravity potential model)
$\lambda(\vec{r})$ or λ	m	Real valued parameter; defines a confocal ellipsoid in the CDE potential model
μ	m^3/s^2	Gravitational parameter
ω	rad/s	Asteroid rotation rate or angular velocity
ω_f	-	Per-facet factor in the polyhedron model
$\vec{\omega}$	rad/s	Angular velocity vector
ϕ_B^I	-	Rotating frame (ARF) to Inertial frame (AIF) transformation matrix

ϕ_I^B	-	Inertial frame (AIF) to Rotating frame (ARF) transformation matrix
ϕ	-	Function in CDE potential model
ρ	m	Distance to a discrete mass distribution
$\vec{\rho}$	m	Distance vector to a discrete mass distribution
σ	kg/m^3	Density in polyhedron model
θ	deg	Rotating angle between the Asteroid-centric inertial and Rotating frame
ϑ	deg	Longitude of Sun with respect to the asteroid

SUBSCRIPTS & SUPERSCRIPTS

Symbol	Description
B	Asteroid-centric Rotating frame (ARF)
e	Edge of a facet in a polyhedron shape model
f,f'	Facet or face of a polyhedron shape model
I	Asteroid-centric Inertial frame (AIF)
l	Degree of spherical harmonics expansion
m	Order of spherical harmonics expansion
n	Degree of ellipsoidal harmonics expansion
p	Order of ellipsoidal harmonics expansion
P	Particle or Regolith
S	Sun
x	Vector component along x-axis
y	Vector component along y-axis
z	Vector component along z-axis
$*_I^B$	AIF ' <i>quantity</i> ' expressed in ARF components
$*_B^I$	ARF ' <i>quantity</i> ' expressed in AIF components

LIST OF ACRONYMS

AIF	Asteroid-Centric Inertial Frame
AOP	Argument Of Periapsis
ARF	Asteroid-Centric Rotating Frame
AU	Astronomical Unit
CCW	Counter-Clockwise
CDE	Constant Density Ellipsoid
CGES	Conservative Guaranteed Escape Speed
CKBO	Classical Kuiper-Belt Object
EHAO	Extremely-High Altitude Orbit
EOM	Equations Of Motion
ESA	European Space Agency
HAO	High Altitude Orbit
HEV	Hyperbolic Excess Velocity
JAXA	Japan Aerospace Exploration Agency
LAF	Low-Altitude Flyover
LAO	Low Altitude Orbit
MAB	Main Asteroid Belt
MAO	Medium Altitude Orbit
MBO	Main-Belt Object
NAOS	Near-Asteroid Orbit Simulator
NASA	National Aeronautics and Space Administration
NCGES	Non-Conservative Guaranteed Escape Speed
NEA	Near-Earth Asteroid
NEAR	Near Earth Asteroid Rendezvous
NEO	Near-Earth Object
OSIRIS-REx	Origins, Spectral Interpretation, Resource Identification, Security, Regolith Explorer
RAAN	Right Ascension of Ascending Node
SCI	Small Carry-on Impactor
SDO	Scattered Disk Object
SF	Surface Frame
SKG	Strategic Knowledge Gap
SRP	Solar Radiation Pressure
STBE	Solar Third Body Effect
TAG	Touch-And-Go
TNO	Trans-Neptunian Object
TUDAT	Technical University Delft Astrodynamics Tool-box
UHAO	Ultra-High Altitude Orbit

1

INTRODUCTION

At the dawn of the nineteenth century, Italian astronomer Giuseppe Piazzi was engrossed in observing the Taurus constellation to update a star catalog. On January 1 1801, atop the Palermo observatory in Sicily, he observed a light which was not mentioned in the catalog. He followed the strange light for a few more nights, eventually realizing that he had discovered a small planet between Mars and Jupiter. He named the minor planet Ceres and it became the first of its kind to be discovered by humans. Broadly speaking, it became the first asteroid to ever be discovered (Cunningham 2016). Soon after this discovery, three other minor planets were discovered in the gap between Mars and Jupiter. Pallas was discovered in 1802, followed by Juno in 1804, and finally Vesta in 1807. After the discovery of Ceres and Pallas, renowned astronomer William Herschel realized that these are a new species of celestial bodies and proposed to call them asteroids (which in Greek means star-like) instead of minor planets. For nearly 40 years after the discovery of Vesta, no additional discoveries were made. Then once again in the second half of the nineteenth century, astronomers started discovering more and more asteroids until they realized that there is a whole belt of it between Mars and Jupiter (Bottke 2002).

Asteroids are rocky, airless celestial bodies in our Solar System that orbit the Sun and are quite small in size compared to the planets. They can be viewed as remnants of the processes that formed the inner planets of our Solar System (NASA 2017a). Asteroids are mostly irregularly shaped with a few exceptions, like Ceres, that have a nearly spherical shape. Figure 1.1 provides a view on the different morphologies of asteroids (NASA 2017a). They are typically categorized based on their location in the Solar System. A large number of asteroids are found in the region between Mars and Jupiter and are called as Main-Belt Objects (MBOs). A relatively smaller number of asteroids, called Near-Earth Asteroids (NEAs), have orbits that are very close to and/or crosses the heliocentric orbit of Earth. Asteroids at the L_4 and L_5 Lagrange points of Jupiter, sharing its orbit around the Sun, are termed as Trojans. Then we have Centaurs, asteroids whose orbit lies between or crosses that of the Giant planets in our Solar System. The fifth and the final category is that of the Trans-Neptunian Objects (TNOs) i.e. asteroids with orbit beyond that of Neptune and reaching as far as the Oort cloud (De Pater et al. 2015). The distribution of asteroids in the inner and outer Solar System is shown in Figure 1.2.

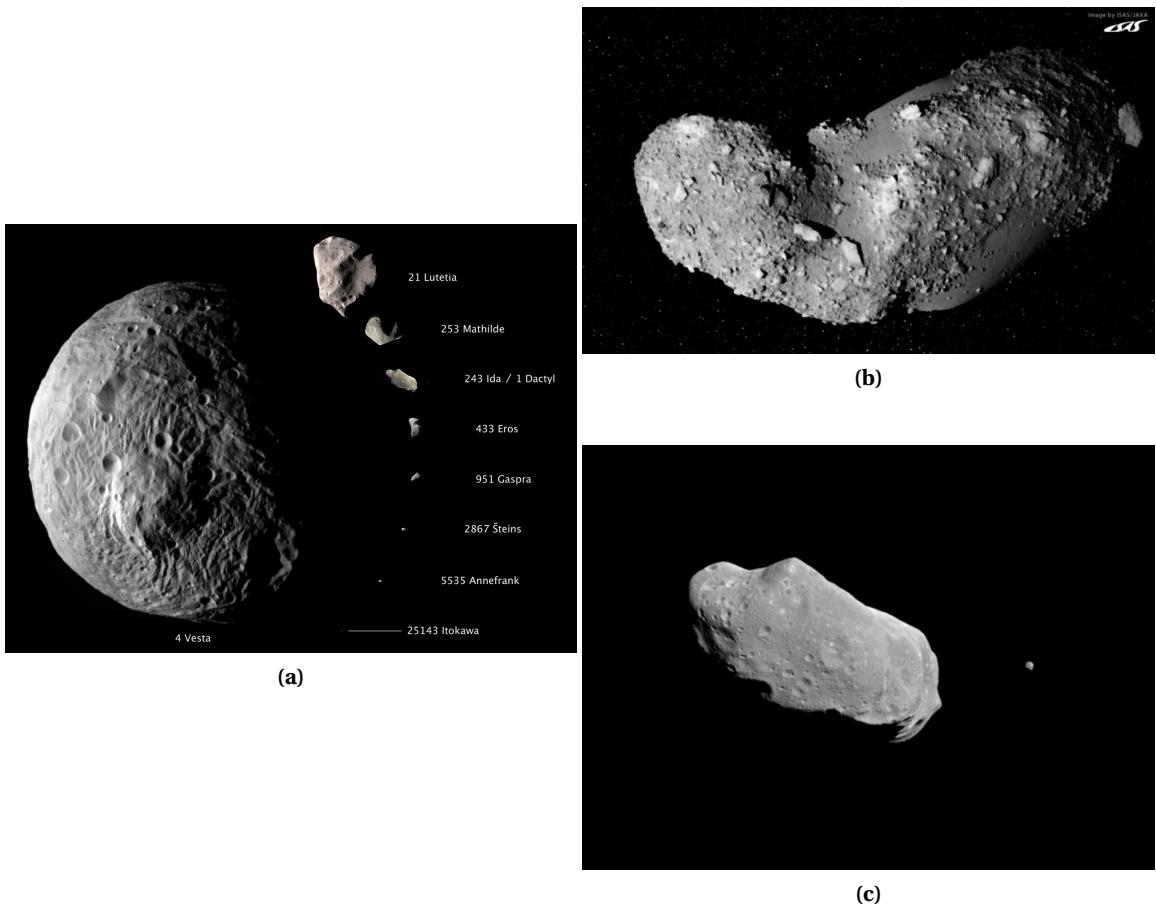


Figure 1.1: Satellite imagery depicting different morphologies of asteroids. (a) Size and shape variations amongst a few known asteroids, (b) Asteroid Itokawa with its rocky and rough surface, (c) Asteroid Ida with its moon Dactyl orbiting around it ([NASA 2017a](#)).

Due to their extremely small sizes, asteroids can not have high internal pressures and temperatures, which means that they could have potentially preserved the early chemistry of our Solar System (Kubota et al. 2006). This makes them a valuable source for us to learn about the history and origin of our Solar System. It is hypothesized that during the early years of Earth's formation, carbon-based molecules and other volatile materials which serve as the basic building-blocks of life, could have been delivered to Earth through asteroid impacts (JPL 2017). Finally, some asteroid types are rich in resources and contain vast supplies of precious metals (Kargel 1994) and water (Morbidelli et al. 2000), which could potentially be mined and used to aid further exploration and colonization of our Solar System (JPL 2017). Thus in light of this, asteroid exploration, both in-situ and ex-situ, has gained significant importance not only among the scientific community but among the private space industry as well, with more and more future missions being planned to these small bodies. The Near Earth Asteroid Rendezvous (NEAR) spacecraft launched by the National Aeronautics and Space Administration (NASA) in 1996, as part of their Discovery program, became the first spacecraft in history to orbit an asteroid (433 Eros) and eventually land on it. The spacecraft spent almost a year around Eros, providing extended and comprehensive observations of surface morphology, shape, internal structure and physical properties of the asteroid (Prockter et al. 2002). The Hayabusa mission (formerly MUSES-C) by the Japan Aerospace Exploration Agency (JAXA) entered into orbit around asteroid Itokawa in 2005 and became the first mission to sample the surface of an asteroid, which was subsequently returned to Earth for analysis in 2010 (Yano et al. 2006). These missions have substantially increased our knowledge about the small bodies in our Solar System.

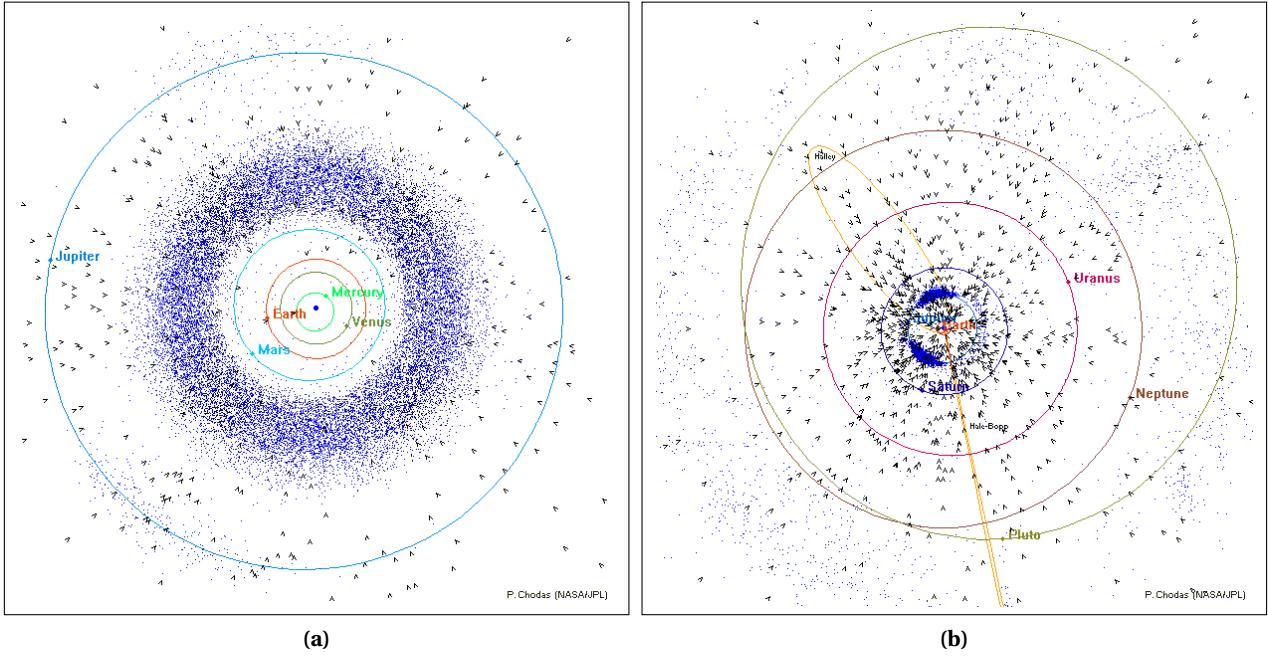


Figure 1.2: Distribution of asteroids in (a) inner Solar System and (b) outer Solar System. Asteroid locations are shown by blue-colored dots whereas the black-colored wedges pointing towards the Sun represent the comets. The diagrams are based on the small-bodies cataloged up until November 2016 ([JPL 2017](#)). These images are color-inverted versions of the original.

Two more asteroid rendezvous missions launched quite recently are of particular interest to this thesis. Following the success of Hayabusa, JAXA launched another sample return mission called Hayabusa-2 to asteroid 1999 JU3, scheduled to be in orbit around it by mid-2018. It will perform a 1.5 year long close-proximity operation at the asteroid that includes surface sample acquisition, which will eventually be returned to Earth in a capsule, and a 2 m wide cratering event to observe the sub-surface (Tsuda et al. 2013). The Origins, Spectral Interpretation, Resource Identification, Security, Regolith Explorer (OSIRIS-REx) mission by NASA, directed towards asteroid Bennu and scheduled to enter into an orbit around it in late 2018, will also retrieve a surface sample and return it back to Earth. It will employ a Touch-And-Go (TAG) maneuver to acquire a sample within a 1.5 month long scheduled sampling period (Berry et al. 2013). Both missions are aiming to find out if organic material, volatiles and water itself were brought to Earth by such asteroids. These missions employ techniques for sample acquisition that could potentially disturb the state of regolith on the surface of the asteroid and loft it into an orbit. Even without any spacecraft interaction, material can be lofted from the surface of an asteroid due to meteoroid impacts. In either case, it is imperative to understand the complex dynamical environment around asteroids, not only for spacecraft operations and safety, but also to learn about the orbital motion of lofted regolith and its eventual fate. NASA has identified the acquisition of such information as a Strategic Knowledge Gap (SKG) for Near-Earth Objects (NEOs), specifically article III-A-1: Expected particulate environment due to impact ejecta in [NASA 2017b](#).

The study of lofted regolith around an asteroid is by no means a new research topic. In the studies done previously (Richter et al. 1995; Lee 1996; Scheeres et al. 1996; Scheeres et al. 2000; Korycansky et al. 2004; Yáñez et al. 2014), we have witnessed certain minor drawbacks such as not always ac-

counting for gravity and Solar perturbations together, or using an approximated analytical method to understand the dynamical environment that falls short on obtaining the entire spectra of initial conditions that could lead to different final outcomes (re-impact, escape or temporary capture) for lofted regolith, or not considering different size and density for the lofted regolith, and finally not considering the local direction with respect to a rotating asteroid in which the regolith is ejected. This thesis, thus, aims to include all of these shortfalls in a single study, and by using numerical simulation techniques, to add more fidelity in understanding what happens to regolith when it is lofted from the surface of an asteroid.

The study of orbiting regolith is important for understanding the displacement of material on surface of the asteroid in case of natural or spacecraft-induced impact cratering events. In case of the latter, the ejecta from the impact cratering event could pose a serious threat to spacecraft and/or its instruments. By knowing the orbital behavior of regolith in advance, mission designers can make informed decisions on the trajectory design of spacecraft to avoid or reduce failure scenarios. Another important benefit that comes from a study like this is in the field of asteroid mining, whereby the regolith's orbital motion and final fate can be exploited to sort different materials in real time. The results from this thesis will thus aid mission designers in planning future asteroid missions and in answering the following research question:

Can we explain the orbital behavior and eventual fate of lofted regolith around an asteroid in presence of gravity and Solar perturbations?

The structure of this report is mentioned as follows: Chapter 2 briefly discusses the past and the future missions to asteroids in the context of this thesis, and a few research publications that are relevant for our work. This paves the way to define the research questions and goals in Chapter 3. Chapter 4 discusses the models for simulating an asteroid's gravity and the Solar perturbations, equations of motion for the regolith and its launch conditions, and finally a non-conservative algorithm to determine a regolith's guaranteed escape speed. The verification and validation of the simulation models can be found in Chapter B. Chapter 5 discusses the performance of the conservative guaranteed escape speed algorithm for a spherical and an ellipsoidal asteroid and the performance of the non-conservative algorithm for the ellipsoidal asteroid. Chapter 6 discusses the general characteristics of lofted regolith and its final fate for an ellipsoidal asteroid in the absence of Solar perturbations. Chapter 7 does the same analysis but in the presence of Solar perturbations and for regoliths of different size and density. Chapter 8 summarizes all the results by presenting answers to all the sub-research questions from Chapter 3. Finally, Chapter 9 provides some recommendations for future work on our research topic.

Part I

Motivation

2

HERITAGE

In the past, there have been multiple spacecraft missions to the small bodies in our Solar System which have collectively increased our understanding about them. While a large majority of these have been asteroid fly-by scenarios, a few have also been rendezvous missions (ESA 2014). This chapter will provide an overview on a few of these missions, followed by a brief literature review which shall be of interest to the thesis at hand. This will help us in justifying the research objectives mentioned in Chapter 3. Section 2.1 will discuss the asteroid rendezvous missions which have already taken place, Section 2.2 will discuss future rendezvous missions, and finally Section 2.3 will discuss the state-of-the-art.

2.1 PAST MISSIONS

In all history of space exploration there have been only three spacecraft missions that have rendezvoused with asteroids. In chronological order these are: NASA's NEAR-Shoemaker mission to asteroid Eros, JAXA's Hayabusa mission to asteroid Itokawa, and NASA's Dawn mission to asteroids Vesta and Ceres (Scheeres 2016). Out of these, only NEAR and Hayabusa had direct contact with the small bodies and acquired high-resolution imagery of surface regolith.

2.1.1 NEAR-SHOEMAKER

The NEAR-Shoemaker (henceforth NEAR) mission was launched in 1996 and rendezvoused with Eros in 2000. Its operational phase around the asteroid continued for about a year during which it obtained several high-resolution images of the surface and collected comprehensive measurements to estimate its internal mass distribution, shape model, gravity and spin state amongst other observations (Scheeres 2016). The bulk density of Eros was estimated to be $2.67 \pm 0.03 \text{ g/cm}^3$ and its mass to be $(6.6904 \pm 0.003) \times 10^{15} \text{ kg}$. The rotation state was estimated to be $1639.38922 \pm 0.00015 \text{ deg/day}$ which gives a rotational period of about 5.27 hours (Miller et al. 2002). On 25 October 2000, NEAR executed a Low-Altitude Flyover (LAF) over Eros in which it acquired several high-resolution images that helped in understanding the surface morphology. The images confirmed the existence of a substantial amount of regolith on the surface with a typical thickness of tens of metres over the bedrock, except on steep slopes. The regolith was found to be highly complex, in that it varied from fine material to metre-sized ejecta blocks (Veverka et al. 2001a). Robinson et al. 2001 estimates the size of the finer regolith to be around 1.0 cm or smaller from images that had a resolution of 1.2 cm per pixel. Figure 2.1 depicts the regolith morphology in one of the high-resolution imaging sequences from the LAF (Veverka et al. 2001b).

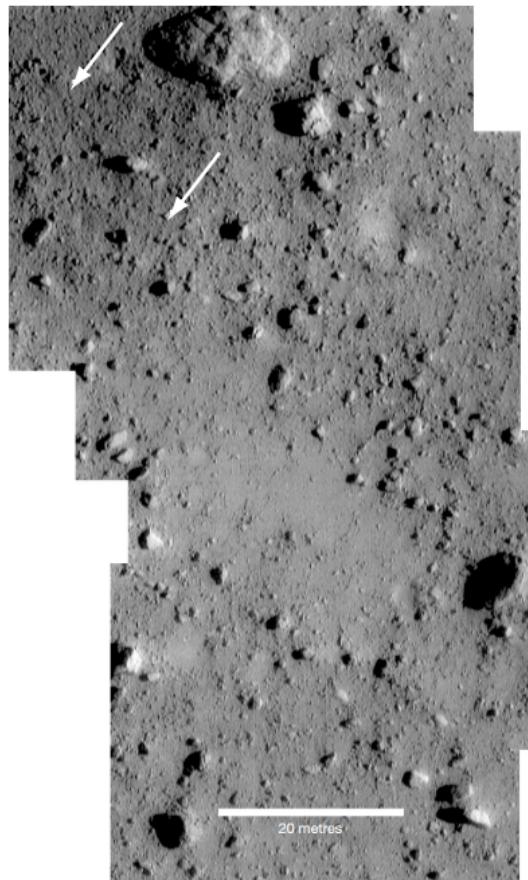


Figure 2.1: Mosaic of high-resolution images depicting the nature of regolith on the surface of Eros. The arrows in the image point to the region for which zoomed-in images are available but are not shown in this report. They can be found in Veverka et al. 2001b.

2.1.2 HAYABUSA

The Hayabusa spacecraft was launched by JAXA in 2003 and it arrived at asteroid Itokawa in 2005. After arrival, it performed close-proximity operations around the asteroid for approximately three months during which several measurements were taken to estimate the shape, mass, topography and elemental composition of the asteroid. During this period, the spacecraft also collected samples from the surface of the asteroid that were eventually returned back to Earth in 2010. The measurements at Itokawa estimated its mass to be 3.51×10^{10} kg and its bulk density to be 1.9 ± 0.13 g/cm³ (Fujiwara et al. 2006).

Two distinct types of terrain can be recognized on Itokawa; one which is rough and rich in boulders and the other which is smooth and mostly flat. This distinction can easily be seen in Figure 2.2. The smooth regolith regions, that account for approximately 20% of Itokawa's surface, are composed of fragmented debris with grain sizes ranging from sub-centimetre to centimetre scales. One of the smooth regolith regions, called Muses Sea and from where the sample was also acquired, even consisted of a few metre-sized boulders that were hypothesized to have landed in the region as secondary ejecta (Miyamoto et al. 2006). The rougher terrain on Itokawa, which has a very sharp boundary with the smoother regolith filled regions (as evident in Figure 2.2), consists of boulders that range up to tens of metres in size (Fujiwara et al. 2006).

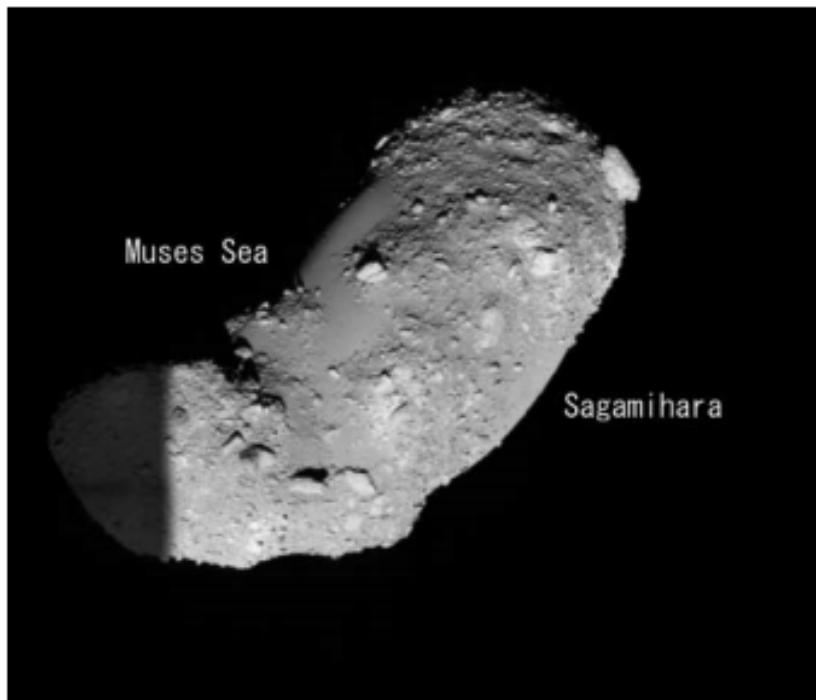


Figure 2.2: Image of Itokawa taken from a 7 km altitude depicting the nature of regolith on its surface. Muses Sea and Sagamihara are two distinct smooth regolith regions on the asteroid (Fujiwara et al. 2006).

Hayabusa employed an impact sampling mechanism that would work across various types of terrains, from hard bedrock to fine regolith. The spacecraft consisted of a long cylindrical sampling horn with a conical tip. When the tip of the horn touched the surface of the asteroid, the deformation in the horn's fabric was detected by a laser range finder and within 0.3 s of this event, a 5.0 g projectile was fired towards the surface with a velocity of 300 m/s and the resultant ejecta was collected by the sampler (Yano et al. 2004). Yano et al. 2006 presents data from the sampling experiments that were performed on ground in 1.0 g and micro-gravity environments. The experiments revealed that, for the projectile hitting at normal impact angles in micro-gravity, the impact ejecta mass of particles greater than 1.0 cm ranged from 2 - 11 g whereas for particles less than 1.0 mm the ejecta mass ranged from 100 - 10000 g. The impact target consisted of various analog materials from glass beads to lunar regolith simulant and an experiment like this is a nice indicator of how artificial impact events can displace significant amount of fragmented debris on an asteroid.

2.2 FUTURE MISSIONS

We will now discuss two missions, Hayabusa-2 by JAXA and OSIRIS-REx by NASA. Both are currently en route to their respective target asteroids and after orbit insertion, they shall perform operations to collect surface samples.

2.2.1 HAYABUSA-2

Hayabusa-2 is the second asteroid sample return mission by JAXA, which to a significant extent shares the successful technical legacy of Hayabusa. The target asteroid of the former is 1999 JU3 which is suspected to contain organic matter and hydrated minerals (Tsuda et al. 2013). A shape model of the asteroid, also designated as Ryugu, is shown in Figure 2.3 (Müller et al. 2017). A successful sample return from this asteroid may thus help us in understanding the origin of life and/or water on Earth. The spacecraft will enter into an orbit around its target by mid-2018, after which it

will perform close-proximity operations for 1.5 years. The mission will entail three touchdowns for sample acquisition and a cratering event to observe the subsurface of the asteroid. The sampling mechanism is based on that of Hayabusa and each sampling attempt has the potential to acquire samples in the order of 100 mg. The samples are sealed-off and transported back to Earth in a re-entry capsule. The cratering operation is performed by a Small Carry-on Impactor (SCI). The SCI is deployed by the spacecraft at an altitude of 500 m and after a preset time, a detonation accelerates it to about 2 km/s prior to impact. It is estimated that this will result in a crater of about 2 m wide. Prior to the detonation of SCI, the spacecraft will move to a safe location on the opposite side of the asteroid from the impact point to avoid damage from impact ejecta and/or debris from the detonation. Apart from these, the spacecraft will perform other in-situ operations to characterize the asteroid and will also deploy a lander and three miniature rovers for technology demonstration (Tsuda et al. 2013).

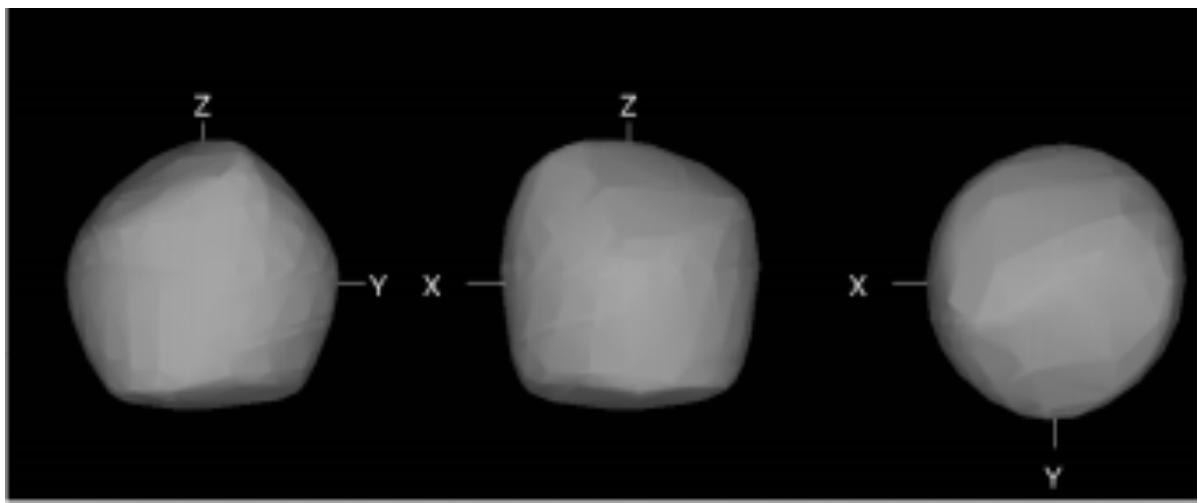


Figure 2.3: Ryugu shape model as estimated from the observations made by Herschel Space Observatory, supported by several ground-based measurements and data from other space-based assets (Müller et al. 2017).

2.2.2 OSIRIS-REX

OSIRIS-REx is part of NASA's New Frontiers program and will travel to NEA 1999 RQ₃₆, also known as Bennu. A shape model of the asteroid is shown in Figure 2.4 (Lauretta et al. 2015). The mission, amongst other scientific objectives, will return a regolith sample back to Earth that may provide insight into the initial states of planetary formation as well as answer questions on the origins of life. Since Bennu is a NEA, the sample collection and subsequent analysis will provide us information on asteroids that could potentially impact Earth. The spacecraft was launched in 2016 and is expected to reach its target by the end of 2018 (Berry et al. 2013). The asteroid has a semi-major axis of 1.126 AU which makes it an easily accessible asteroid as far as distance is concerned. But more than that, Bennu falls under the category of asteroids that are rich in volatiles and could potentially be related to objects that brought the seeds of life to Earth. Initial observations of Bennu through ground-based telescopes, the Spitzer Telescope, the Arecibo Observatory and other assets revealed an abundance of regolith on the surface with grain sizes ranging from 4 - 8 mm. OSIRIS-REx will acquire the regolith sample using a TAG mechanism which uses pressurized nitrogen gas to force the loosely-held regolith into a collection chamber. The sampling will occur in 2020 and it will be retrieved on Earth in 2023 (Lauretta et al. 2012).

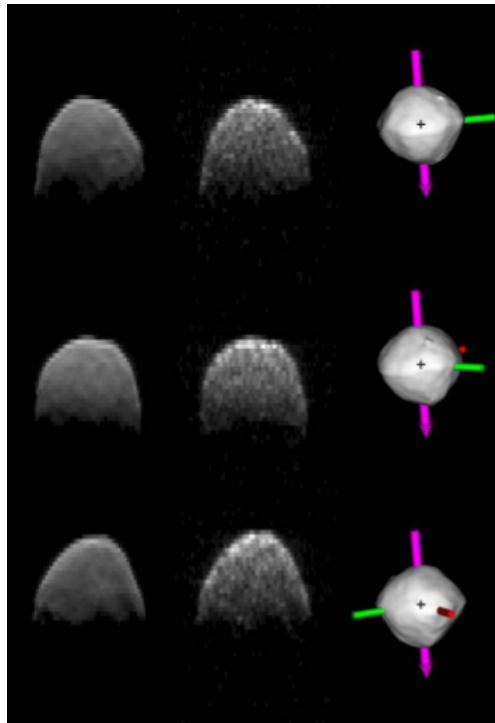


Figure 2.4: Bennu's shape as observed from the radar data collected by the Goldstone and Arecibo observatories (shown in middle column of the image). The left column displays the model that provides the best fit to the radar data and the right column shows the final estimated 3D model of Bennu as it would appear in the sky (Lauretta et al. 2015).

2.3 STATE OF THE ART / LITERATURE REVIEW

In this section we shall discuss a few research papers relevant to this thesis; the techniques they applied to understand the orbital behavior of impact ejecta and the shortcomings of these studies.

A good starting point to understand the topic at hand is provided by Scheeres et al. 2002. It reviews the gravity and perturbing force models along with the dynamical equations of motion for a particle in orbit around an asteroid and the model for generating initial conditions to launch ejecta from the surface of an asteroid. It also mentions existing analytical methods to compute guaranteed escape and re-impact speeds for impact ejecta i.e. speeds at which particles would immediately escape or re-impact after being launched. Scheeres et al. 2002 also discusses the various numerical and analytical methods that have been used in literature for analyzing the motion of particles that stay in pseudo-stable orbits for extended periods of time before meeting their final fate. It also presents various mechanisms that have been hypothesized for the capture-case scenario i.e. particles that stay in orbit around asteroids for a relatively long time, from hundreds of days to several years. The analysis of these capture orbits, in particular, has been done by considering the Solar perturbations and irregular gravity effects of the asteroid but always in isolation.

Richter et al. 1995 provides an analytical method to solve for the motion of particles around a non-rotating, spherical cometary nucleus which is in an eccentric, heliocentric orbit. In their paper, they ignore Solar tidal effects and assume that the particle motion around a homogeneous spherical body would experience weak perturbations from Solar Radiation Pressure (SRP). They give averaged equations for the variation of eccentricity and angular momentum vectors as a function of the true anomaly of the comet around the Sun. The paper also discusses the limitations and validity of using their analytical approximation as well as the conditions for collision-free orbits for small and large dust particles around the comet. Although the study conducted by Richter et al. 1995 is for comets,

it can be extended to asteroids as well and has been used by Morrow et al. 2001 for analyzing solar sail powered trajectories around them. Lee 1996 discusses the electrostatic levitation of dust particles from the surface of an asteroid. It uses two electrostatic field production methods used in the study of dust levitation on a moon, and applies them to the case of an asteroid. The study does not involve the orbital motion of dust particles but it does provide conditions which could cause the dust particle to escape in the event of electrostatic levitation.

Scheeres et al. 1996 provides an extremely detailed and systematic study of particle dynamics close to the surface of asteroid Castalia. They include the effect of the irregular shape of the asteroid on orbital dynamics by using a spherical harmonics model of degree and order upto four in simulating the gravity potential. They also derive analytical results for the computation of guaranteed return and escape speeds as a function of the location of a particle on the surface of the asteroid. The paper employs dynamical systems theory and investigates the use of stable manifolds associated with orbits around equilibrium points and intersecting the surface of the asteroid, to obtain the initial launch conditions for the particle that will lead to a temporary stable orbit around the asteroid. Scheeres et al. 2000 applies the radiation pressure approximation method developed by Richter et al. 1995 to study the temporary capture of particles in an orbit around a comet, but improves it to account for the comet's rotation as well. The results obtained from the analytical approximation are compared with the results from the numerical simulation wherein the latter accounts for other perturbations as well such as Solar tidal effects and gravity field variations. The comparison showed that the radiation pressure approximation method by Richter et al. 1995 is qualitatively correct and can be used for statistical studies at the very least. They were also able to establish qualitative ranges on ejecta velocity and angles that result in capture orbits, an example of which is shown in Figure 2.5.

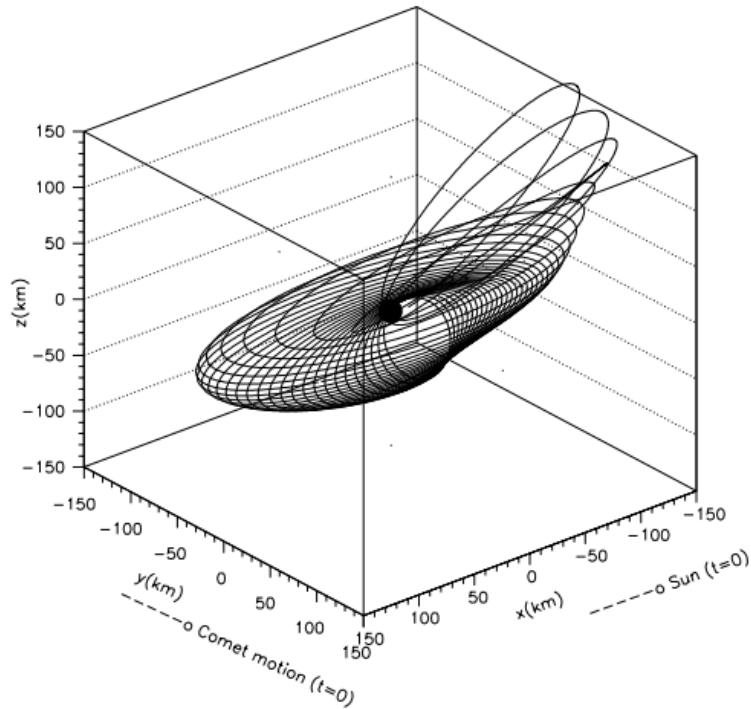


Figure 2.5: Example of a single particle capture trajectory around comet Tempel-1 (Scheeres et al. 2000).

Korycansky et al. 2004 conducts a study to understand the distribution of impact ejecta and its connection with existing regolith on the surface of asteroid 433 Eros. The study involves the use of a Monte Carlo simulation technique to observe the orbital evolution of a large number of test particles

from randomly selected locations on the asteroid. They use a coarse polyhedron model of asteroid Eros to model its gravitational field, thus accounting for gravity perturbations. However, the research does not account for perturbations from SRP. Yáñez et al. 2014 studies the orbital motion of lofted regolith in the context of using SRP to passively sort asteroid material. They use semi-analytic methods to derive conditions that would cause regolith to either escape or re-impact the asteroid's surface. They make use of the radiation pressure approximation methodology developed by Richter et al. 1995 in their semi-analytical approach. However, the effect of an irregular shape of an asteroid, i.e. gravity perturbations, is not accounted for in their calculations.

In general, we witnessed minor drawbacks in these studies such as not always accounting for gravity and Solar perturbations together, or the derivation of an analytical solution which is not globally valid. Some studies involved both analytical and numerical methods for simulating orbital dynamics but even then the numerical approach was more for comparing the validity of the analytical solution and not as much for obtaining the full range of initial conditions that will lead to re-impact, escape or temporary capture of regolith around an asteroid. The effect of launch direction of regolith was also not considered in most of the studies, especially the ones that applied analytical methods. We attempt to address these shortfalls to better understand the reasons for the complex orbital behavior of particles launched from the surface of an asteroid, by following a numerical simulations approach instead of an analytical approximation or a dynamical theory one (see Scheeres et al. 2002 for a brief discussion between the three methods for analyzing orbital behavior of asteroid ejecta). We account for gravity and Solar perturbations while simulating trajectories for particles of different sizes and density. These perturbations have been considered in isolation as well as together to witness the effect of each individual perturbation on a particle trajectory.

3

RESEARCH QUESTIONS AND GOALS

The study of the dynamics of a particle, on or around an asteroid, can be broadly divided into three main regimes. The first regime involves the study of surface ejecta generation, from natural events such as particle impacts, cratering by other asteroids and electrostatic dust levitation, or from space exploration events where the natural state of the regolith is disturbed by spacecraft sampling activities. The second regime involves the study of the subsequent orbital behavior of impact ejecta or lofted regolith under varying parameters such as launch conditions, asteroid rotational state and shape, regolith particle size and density, Solar phase etc. And finally, the third regime involves the study of particle dynamics when it re-impacts the surface of the asteroid. This thesis will concern itself with the second regime of research, i.e., the natural orbital evolution of regolith lofted from an asteroid's surface, and with a small part of the third regime as well where we study variations in regolith re-impact patterns.

As mentioned earlier in Chapter 1, understanding particulate environment around small bodies has been identified by NASA as a strategic knowledge gap. Understanding and developing tools or knowledge to estimate the orbital behavior and final fate of lofted regolith with greater accuracy is important for future space exploration missions (see Section 2.2) that will involve direct interactions with asteroids, to avoid any damage to the spacecraft or surface robotic crew from orbiting particles. High-fidelity simulations of particulate motion can also help scientists in understanding the surface morphology of asteroids by helping them recreate cratering events. In Section 2.3, we highlighted the shortfalls in the research done on the topic so far and we identified a gap that needs to be filled, and hence, the following top level research question is set:

Can we explain the orbital behavior and eventual fate of lofted regolith around an asteroid in presence of gravity and Solar perturbations?

This top-level research question is divided into the following sub-questions that help in structuring the thesis:

1. Does the regolith, launched from different locations such as leading, trailing, longest edge of an asteroid, show characteristic differences with regard to its final fate?
2. Can we establish a non-conservative analytical expression to determine guarantee escape speed in the presence of perturbations?
3. What causes the regolith to enter into a temporary capture orbit around the asteroid?

4. For the same launch conditions, how does the orbital behavior and final fate of the regolith differ for different particle sizes and densities?
5. Can we exploit the orbital behavior of lofted regolith for sorting material of different sizes and densities as an application for asteroid mining?

In order to answer these questions, the following main research goal is set:

Investigate the orbital motion of regolith launched from the surface of an asteroid using numerical simulations.

The sub-research goals are mentioned as follows:

1. Develop a modular and robust software tool that can propagate the trajectory of spherical particles around an asteroid for given initial conditions.
2. Develop software tools to plot and analyze numerical simulation results.
3. Validate the software tools.
4. Perform simulations for particles launched from the asteroid's surface with different initial conditions, launch locations, and for different particle sizes and densities.
5. Perform qualitative and quantitative analysis on numerical simulation results.
6. Document results and inferences for this thesis report and a peer-reviewed journal paper.

Part II

Dynamics Modeling

4

ORBITAL DYNAMICS AROUND ASTEROIDS

This chapter will focus on accurate modeling of the asteroid environment and the equations of motion of a particle around it in presence of gravity and Solar perturbations.

4.1 MODELING ASSUMPTIONS

The simulator¹ designed as part of this research involves some degree of approximation of the real-world dynamical environment around the asteroid. Every degree of freedom and complexity added to a simulator to resemble the real-world, will also act as a potential source of error. By designing a relatively simple simulator, we can explain the characteristic behavior of regolith through fundamentals while keeping the sources of error as low as possible. Of course, we verify the simulator (see Chapter B for a detailed verification and validation process) but by including a higher degree of fidelity in the simulator, we increase the workload on the verification process as well, thereby reducing the scientific output in the end. Moreover, the current simulator and the results from it will act as a benchmark for a higher-fidelity simulator in the future. Thus, the approximations made in this thesis are as follows:

1. The asteroid body is modeled as a smooth triaxial ellipsoid to account for its non-uniform gravity. The triaxial model is chosen over the spherical and ellipsoidal harmonics approach because we want to study the motion of regolith close to the surface of the asteroid; in particular the re-impact scenarios. This can not be done with the harmonics model as the gravity potential diverges, within the circumscribing volume, from the true potential of an irregular body. We chose to not use a polyhedron model either, even though it can account for surface irregularities of an irregular body much better than the triaxial ellipsoid. This decision was made to reduce the complexity of the simulator, however, we do not sacrifice on the qualitative aspect of the results since the triaxial model also provides a non-uniform gravity field.
2. Craters, surface depressions, mountains and any other terrain deformity on the asteroid are not considered in the simulation. The body is considered to have a uniform density. This is to simplify calculations of the gravity acceleration.
3. The asteroid is rotating uniformly about its shortest axis. This is considered for simplicity and also because most Solar System bodies would dissipate energy to eventually enter a rotational state that is uniform and about its axis of maximum moment of inertia (Scheeres 2016). Hence, the approximation for the asteroid remains valid.

¹The simulator is called Near-Asteroid Orbit Simulator (NAOS) and can be downloaded from <https://github.com/agrawalabhishek/NAOS/tree/devcode2>

4. The regolith grains are assumed to be spherical in shape to simplify the SRP calculation as the cross-sectional area of a sphere would remain the same irrespective of its attitude. In addition to this, the grains are assumed to have an albedo equal to one, which means that any impinging Solar photons are completely reflected and this is done to account for maximum perturbation from SRP for a given location and area-to-mass ratio (see Section 4.4.2 for more details).
5. Multiple regolith particles are launched from a given location on the asteroid in the form of a cone to replicate ejecta from a cratering event. But all particles are assumed to be coming off from the same point, unlike that in the case of an actual cratering event. This is because the pretext of the thesis was that the regolith is lofted due to an activity from a spacecraft which would result in relatively smaller craters or surface depressions. Thus assuming that all particles in this "*ejecta cone*" emerge from the same point on the asteroid is reasonable and simplifies the simulation.
6. The slant angle of the "*ejecta cone*" (henceforth the declination angle) from the local surface normal is kept constant at 45.0° (which is a middle value in the entire declination range from 0.0° - 90.0°). We want to consider a general case and not introduce another degree-of-freedom in terms of varying declination angles, however, it should be noted that the results from simulation for any other declination angle would be different. There is an exception to the fixed declination angle assumption though; only in Chapter 5, we use a range of declination angles at which regolith is launched.
7. The calculations were subjected to simulate a maximum of 270 days and were terminated earlier if a particular trajectory resulted in escape or surface re-impact. This number was obtained by looking at the close-proximity operational time periods of exploration missions to small bodies of our Solar System. We wanted a maximum simulation time in the context of a man-made mission and hence this approach was taken. We accounted for four missions, two from the past and two planned for the future, which have direct contact with a small body as part of their mission and continued the mission around the small body afterwards (hence, not just disposal and/or fly-by). These are the Philae (Rosetta), Hayabusa, Hayabusa-2, and the OSIRIS-REx missions. The close-proximity design operation time period for the Philae lander was three months (Biele et al. 2008), three months for Hayabusa (Kawaguchi et al. 2003), eighteen months for the Hayabusa-2 mission (Tsuda et al. 2013), and finally twelve months for the OSIRIS-REx mission (Lauretta et al. 2012). The average of all of this comes out to be nine months, which is what we have considered to be the maximum simulation time. In this regard, we are also categorizing orbital behavior that does not result in escape or re-impact in those 270 days, as capture orbits.
8. The loss of material and mass from the asteroid, when the regolith is lofted from the surface, is not modeled in the simulation since it is assumed that a very small amount of material will be displaced by a spacecraft activity (impact and/or sampling). This assumption is based on the sample collected by the Hayabusa mission (see Section 2.1 and the references therein).
9. Interaction between individual regolith grains is not accounted for because we are simulating multiple particles being lofted at the same time and granular interaction on such a scale would be extremely complex and small and beyond the scope of this thesis.
10. Secondary motion of regolith, after re-impacting the surface, is not modeled and it is assumed that the particles just come to a standstill.

11. The shadow region of the asteroid is not modeled which means that the solar perturbations are always acting on the regolith grain. This simplification was made since asteroids are extremely small compared to planets, and thus, the chances of an orbiting particle spending long periods of time in the shadow are very small.
12. Perturbations are considered only from the Sun. SRP is important because regolith grains will have higher area-to-mass ratios, relative to a spacecraft, and so the radiation pressure would be significantly large for them. We model the third-body attraction from the Sun (Solar Third Body Effect (STBE)) as well but not from any of the planets because we are assuming that the small body does not pass close to any planet, thus rendering the perturbations from them insignificant.
13. The apparent motion of the Sun around the asteroid is considered circular and in the equatorial plane of the asteroid, again, to restrict another degree of freedom in the simulations and consider a general case.

4.2 REFERENCE FRAMES

Before describing the motion of regolith around the asteroid, it is important to define the frames of reference with respect to which this motion is defined and the transformation of state vectors between these frames. We use two asteroid-centric reference frames, both of which are depicted in Figure 4.1. Since we will be using a triaxial ellipsoid to model an asteroid (for details, see Section 4.3), the body-fixed rotating frame and the inertial frame with respect to this model are shown in Figures 4.2 and 4.3 respectively.

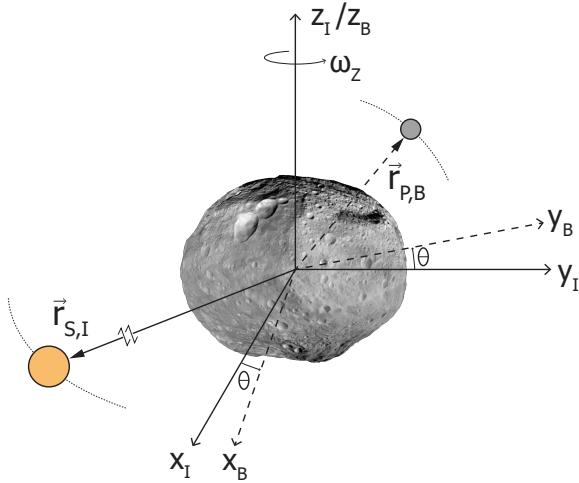


Figure 4.1: Two asteroid-centric reference frames, one being inertial (depicted by solid lines and the subscript I) and the other being a body-fixed rotating frame (denoted by dashed lines and the subscript B). The position vector to a regolith particle is shown as $\vec{r}_{P,B}$, whereas the position vector to the Sun is shown as $\vec{r}_{S,I}$.

The two frames are defined as follows:

1. Asteroid-Centric Inertial Frame (AIF) - This is a non-rotating frame fixed inertially in space with its origin at the centre of mass of the asteroid. Figure 4.2 shows the orientation of the frame (in x-y plane) such that the x-axis is pointing to the Sun when the longitude of the Sun (or effectively the true anomaly of the apparent circular motion of the Sun around the asteroid) ϑ is zero. The y-axis, thus, points to the Sun when $\vartheta = 90^\circ$ and finally the z-axis is obtained by following the right-hand rule, coming out of the sheet in 3D.
2. Asteroid-Centric Rotating Frame (ARF) - This frame is fixed to the rotating asteroid with its origin at the centre of mass of the asteroid and its axes aligned with the principle axes of the asteroid. Figure 4.3 shows the orientation of this frame, assuming a triaxial ellipsoid model (for details see Section 4.3). The x-axis is pointing along the longest axis of the triaxial ellipsoid, and the z-axis is pointing along the shortest axis. It is also aligned with the z_I -axis of the AIF as shown in Figure 4.1. The y-axis points in the direction of the remaining third axis, satisfying the right-hand rule. The asteroid (and effectively the ARF) is rotating in a counter-clockwise sense with respect to the AIF, with constant angular velocity ω about the z_B -axis as depicted in Figure 4.1.

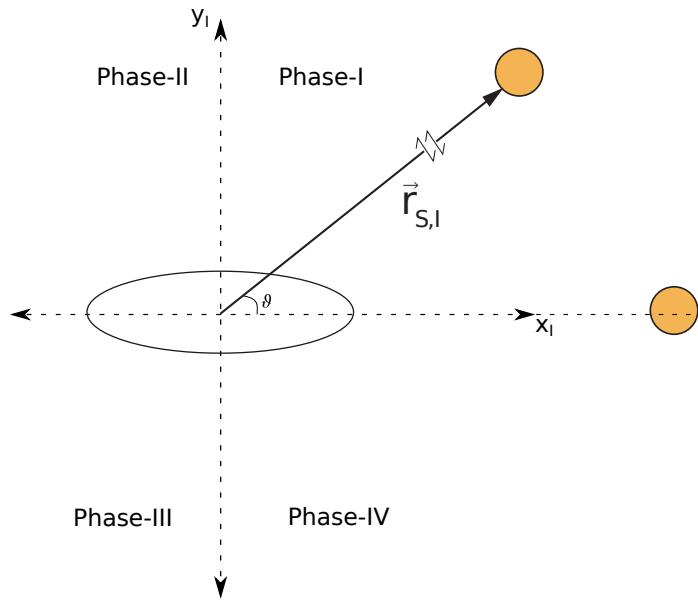


Figure 4.2: Asteroid-centric inertial frame x - y plane. The position vector to the Sun is shown as $\vec{r}_{S,I}$ with ϑ as the longitude of the Sun (or effectively the true anomaly). The four phases are for a broader identification of the Sun's location with respect to the asteroid.

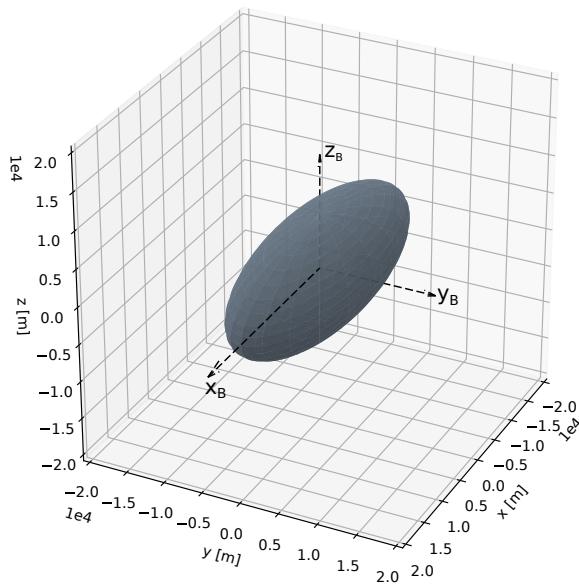


Figure 4.3: Representation of the body-fixed rotating frame for a triaxial ellipsoid model of an asteroid. x_B is aligned with the longest axis, z_B is aligned with the shortest axis and y_B is aligned with the remaining last axis of the ellipsoid, satisfying the right-hand rule. The asteroid rotates uniformly about the z_B -axis.

We have two different frames of reference because it is important to visualize the same orbital motion with respect to both an inertial frame and a non-inertial frame to get a better understanding of the underlying dynamics. In this regard, it is thus important to be able to transfer a state vector between the two frames. The transfer matrix to transform a state vector from the ARF to the AIF is given as follows (Schaub et al. 2003):

$$\phi_B^I = \begin{bmatrix} \cos\theta & -\sin\theta & 0 \\ \sin\theta & \cos\theta & 0 \\ 0 & 0 & 1 \end{bmatrix} \quad (4.1)$$

$$\theta = \omega t$$

In Equation (4.1), θ is the angle of rotation between the ARF and the AIF at any given time t ; and ω is the constant angular velocity of the rotating asteroid about the z_I/z_B -axis as shown in Figure 4.1. The position vector is then transformed, from ARF to AIF, as follows (Schaub et al. 2003):

$$\begin{bmatrix} x_I \\ y_I \\ z_I \end{bmatrix} = \phi_B^I \begin{bmatrix} x_B \\ y_B \\ z_B \end{bmatrix} \quad (4.2)$$

In Equation (4.2), (x_I, y_I, z_I) and (x_B, y_B, z_B) are the components of the position vector in the AIF and the ARF respectively. The velocity transformation takes place by first using the *transport theorem* and then multiplying the resultant with the transformation matrix ϕ_B^I (Schaub et al. 2003). The transformation is shown as follows:

$$\vec{v}_I^B = \vec{v}_B + \vec{\omega} \times \vec{r}_B \quad (4.3)$$

$$\vec{v}_I = \phi_B^I \vec{v}_I^B \quad (4.4)$$

Equation (4.3) is the application of the transport theorem to get the AIF velocity in ARF components (\vec{v}_I^B). In that, \vec{v}_B is the velocity vector in the ARF, $\vec{\omega}$ is the angular velocity vector for the asteroid's rotation (note that we have only uniform rotation about the z_B -axis), and \vec{r}_B is the position vector defined in the ARF. In Equation (4.4), \vec{v}_I is the velocity vector in the AIF.

The transfer matrix to transform a state vector from the AIF to the ARF is just the transpose of ϕ_B^I as it is orthogonal (Schaub et al. 2003). It is given as follows:

$$\phi_I^B = \begin{bmatrix} \cos\theta & \sin\theta & 0 \\ -\sin\theta & \cos\theta & 0 \\ 0 & 0 & 1 \end{bmatrix} \quad (4.5)$$

Then the state vector transformation from AIF to ARF takes place as follows:

$$\vec{r}_B = \phi_I^B \vec{r}_I \quad (4.6)$$

$$\vec{v}_B^I = \vec{v}_I - \vec{\omega} \times \vec{r}_I \quad (4.7)$$

$$\vec{v}_B = \phi_I^B \vec{v}_B^I \quad (4.8)$$

where \vec{v}_B^I is the ARF velocity in AIF components, and \vec{r}_I is the position vector in the AIF.

4.3 GRAVITY POTENTIAL

The key feature that differentiates small bodies, or asteroids for our particular case, from planets is their highly irregular shapes and thus non-spherical mass distributions (Scheeres 2016). This is why the dynamics close to an asteroid are deemed as interesting and hence it is very important that the gravity potential is modeled properly. We will only discuss the CDE model in this chapter since we chose to work with it, however, a small discussion on the spherical and ellipsoidal harmonics and the constant-density polyhedron models are presented in Section A.1 and Section A.2 respectively.

4.3.1 CONSTANT DENSITY ELLIPSOID

Consider a CDE with semi-axes (α, β, γ) such that $\gamma \leq \beta \leq \alpha$. The shape of the triaxial ellipsoid is completely defined by the equation $(x/\alpha)^2 + (y/\beta)^2 + (z/\gamma)^2 = 1$. The density of the ellipsoid is assumed to be constant. An example for a CDE model is shown in Figure 4.4. The gravity potential for a point external to such a body, i.e. CDE, is defined by the following equations (Scheeres 2016):

$$U(\vec{r}) = -\frac{3\mu}{4} \int_{\lambda(\vec{r})}^{\infty} \phi(\vec{r}, u) \frac{du}{\Delta(u)} \quad (4.9)$$

$$\phi(\vec{r}, u) = \frac{x^2}{\alpha^2 + u} + \frac{y^2}{\beta^2 + u} + \frac{z^2}{\gamma^2 + u} - 1 \quad (4.10)$$

$$\Delta(u) = \sqrt{(\alpha^2 + u)(\beta^2 + u)(\gamma^2 + u)} \quad (4.11)$$

where \vec{r} is the position vector to the point, external to the CDE, and is defined in the ARF; $\lambda(\vec{r})$ is a parameter defined by the equation $\phi(\vec{r}, \lambda) = 0$, which is a cubic polynomial as shown in Equation (4.12), and the value λ is the maximum real root of this polynomial (Scheeres 2016).

$$\begin{aligned} & \lambda^3 + \\ & \lambda^2(\alpha^2 + \beta^2 + \gamma^2 - (x^2 + y^2 + z^2)) + \\ & \lambda(\alpha^2\beta^2 + \alpha^2\gamma^2 + \beta^2\gamma^2 - x^2(\beta^2 + \gamma^2) - y^2(\alpha^2 + \gamma^2) - z^2(\alpha^2 + \beta^2)) + \\ & (\alpha^2\beta^2\gamma^2 - x^2\gamma^2\beta^2 - y^2\alpha^2\gamma^2 - z^2\alpha^2\beta^2) = 0 \end{aligned} \quad (4.12)$$

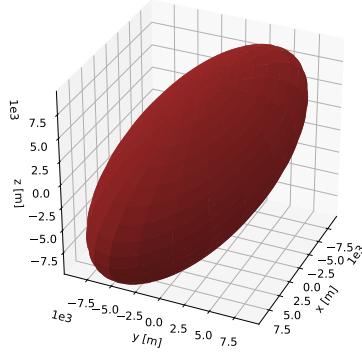


Figure 4.4: Triaxial ellipsoid model with semi-axes $\alpha = 20$ km, $\beta = 7$ km, $\gamma = 7$ km.

For a given point (x, y, z) in space around the CDE, the only unknown in Equation (4.12) is λ which is solved for using the standard Cardano's formula (see Weisstein 2017). For the given ellipsoid, the equation for the family of confocal quadratic surfaces is given as follows (Panou 2014):

$$\frac{x^2}{\alpha^2 + u} + \frac{y^2}{\beta^2 + u} + \frac{z^2}{\gamma^2 + u} = 1 \quad (4.13)$$

where u is a real-valued parameter whose value defines the type of the confocal quadratic surface. Equation (4.13) is a cubic polynomial in u and can be solved to obtain three unequal real roots u_1, u_2, u_3 , such that the following relation holds true (Panou 2014):

$$-\alpha^2 < u_3 < -\beta^2 < u_2 < -\gamma^2 < u_1 < +\infty \quad (4.14)$$

where u_1 is the maximum real root possible and at that value, Equation (4.13) defines another ellipsoid which is confocal to the original one defined by the semi-axes α, β, γ (Panou 2014). Thus, the value of λ in Equation (4.9) conforms to a confocal ellipsoid for a given point external to the original ellipsoid (which in turn is modeling the asteroid).

The potential defined by Equation (4.9) appears to have a complicated computational process due to its integral form. However, the integral can be split into multiple parts such that each can be solved with the help of standard functions called *Carlson's Elliptic Integrals* (Carlson 1987). Software routines for these integrals exist in several computing languages which, for our case, helps in computing the CDE gravity potential. The integral defined in Equations (4.9) to (4.11) is restated in its complete form as follows:

$$U(\vec{r}) = -\frac{3\mu}{4} \int_{\lambda(\vec{r})}^{\infty} \left(\frac{x^2}{\alpha^2 + u} + \frac{y^2}{\beta^2 + u} + \frac{z^2}{\gamma^2 + u} - 1 \right) \frac{du}{\sqrt{(\alpha^2 + u)(\beta^2 + u)(\gamma^2 + u)}} \quad (4.15)$$

Equation (4.15) can be split into four parts which are as follows:

$$U_1 = -\frac{\mu}{2} \cdot \frac{3}{2} \int_{\lambda}^{\infty} \frac{x^2}{(\alpha^2 + u)^{3/2} (\beta^2 + u)^{1/2} (\gamma^2 + u)^{1/2}} du \quad (4.16)$$

$$U_2 = -\frac{\mu}{2} \cdot \frac{3}{2} \int_{\lambda}^{\infty} \frac{y^2}{(\alpha^2 + u)^{1/2} (\beta^2 + u)^{3/2} (\gamma^2 + u)^{1/2}} du \quad (4.17)$$

$$U_3 = -\frac{\mu}{2} \cdot \frac{3}{2} \int_{\lambda}^{\infty} \frac{z^2}{(\alpha^2 + u)^{1/2} (\beta^2 + u)^{1/2} (\gamma^2 + u)^{3/2}} du \quad (4.18)$$

$$U_4 = +\frac{\mu}{2} \cdot \frac{3}{2} \int_{\lambda}^{\infty} \frac{du}{(\alpha^2 + u)^{1/2} (\beta^2 + u)^{1/2} (\gamma^2 + u)^{1/2}} \quad (4.19)$$

where $\lambda(\vec{r})$ is simply written as λ for brevity. Thus, the CDE potential is given as $U = U_1 + U_2 + U_3 + U_4$. We make the following substitution for Equations (4.16) to (4.19):

$$u = v + \lambda \quad (4.20)$$

$$du = dv \quad (4.21)$$

$$u = \lambda; v = 0 \quad (4.22)$$

$$u = \infty; v = \infty \quad (4.23)$$

With these substitutions, Equation (4.16), for example, can now be re-written as follows:

$$U_1 = -\frac{\mu x^2}{2} \left[\frac{3}{2} \int_0^{\infty} \frac{dv}{((\alpha^2 + \lambda) + v)^{3/2} ((\beta^2 + \lambda) + v)^{1/2} ((\gamma^2 + \lambda) + v)^{1/2}} \right] \quad (4.24)$$

$$= -\frac{\mu x^2}{2} R_D(\beta^2 + \lambda, \gamma^2 + \lambda, \alpha^2 + \lambda) \quad (4.25)$$

In Equation (4.24), the expression within the square brackets conforms to the standard elliptic integral function R_D as defined by Carlson 1987 and is given as follows:

$$R_D(x, y, z) = \frac{3}{2} \int_0^{\infty} \frac{dt}{(t + x)^{1/2} (t + y)^{1/2} (t + z)^{3/2}} \quad (4.26)$$

Similarly, Equations (4.17) and (4.18) can be re-written using the standard elliptic integral function R_D as follows:

$$U_2 = -\frac{\mu y^2}{2} R_D(\alpha^2 + \lambda, \gamma^2 + \lambda, \beta^2 + \lambda) \quad (4.27)$$

$$U_3 = -\frac{\mu z^2}{2} R_D(\alpha^2 + \lambda, \beta^2 + \lambda, \gamma^2 + \lambda) \quad (4.28)$$

For Equation (4.19), we use another standard elliptic integral function as defined by Carlson 1987:

$$R_F(x, y, z) = \frac{1}{2} \int_0^\infty \frac{dt}{(t+x)^{1/2}(t+y)^{1/2}(t+z)^{1/2}} \quad (4.29)$$

using which, Equation (4.19) is re-written as follows:

$$U_4 = \frac{3\mu}{2} R_F(\alpha^2 + \lambda, \beta^2 + \lambda, \gamma^2 + \lambda) \quad (4.30)$$

Thus, to calculate the CDE gravity potential U at any given point (x, y, z) external to the ellipsoid, we first calculate the corresponding value of λ from Equation (4.12) and then substitute this value into Equations (4.25), (4.27), (4.28) and (4.30), the sum of which is the final potential value.

The gravity acceleration components are obtained by taking the partial derivatives of the potential equation given in Equation (4.15) (Scheeres 2016):

$$U_x = -\frac{3\mu x}{2} \int_{\lambda(\vec{r})}^\infty \frac{du}{(\alpha^2 + u)\Delta u} \quad (4.31)$$

$$U_y = -\frac{3\mu y}{2} \int_{\lambda(\vec{r})}^\infty \frac{du}{(\beta^2 + u)\Delta u} \quad (4.32)$$

$$U_z = -\frac{3\mu z}{2} \int_{\lambda(\vec{r})}^\infty \frac{du}{(\gamma^2 + u)\Delta u} \quad (4.33)$$

where U_x, U_y, U_z are the gravity acceleration terms and all other terms have the same definition as explained before for the CDE potential term. Just like with the gravity potential, the acceleration terms can be reduced to standard Carlson's elliptic integrals by using the same substitution parameters as defined in Equations (4.20) to (4.23). After substitution, for example, the x-component of the acceleration term is written as follows:

$$U_x = -\frac{-3\mu x}{2} \int_0^\infty \frac{dv}{(\alpha^2 + v + \lambda)\sqrt{(\alpha^2 + v + \lambda)(\beta^2 + v + \lambda)(\gamma^2 + v + \lambda)}} \quad (4.34)$$

$$= -\mu x \left[\frac{3}{2} \int_0^\infty \frac{dv}{((\alpha^2 + \lambda) + v)^{3/2}((\beta^2 + \lambda) + v)^{1/2}((\gamma^2 + \lambda) + v)^{1/2}} \right] \quad (4.35)$$

$$= -\mu x \cdot R_D((\beta^2 + \lambda), (\gamma^2 + \lambda), (\alpha^2 + \lambda)) \quad (4.36)$$

Similarly, the other components of the gravity acceleration (defined in the ARF) can be written as follows:

$$U_y = -\mu y \cdot R_D((\alpha^2 + \lambda), (\gamma^2 + \lambda), (\beta^2 + \lambda)) \quad (4.37)$$

$$U_z = -\mu z \cdot R_D((\alpha^2 + \lambda), (\beta^2 + \lambda), (\gamma^2 + \lambda)) \quad (4.38)$$

For any point (x, y, z) outside of the CDE, we calculate the value for λ first by solving Equation (4.12) and then substitute it into Equations (4.36) to (4.38) to get the acceleration components in the ARF.

4.4 SOLAR PERTURBATIONS

The dominant force acting on an orbiting particle in the vicinity of an asteroid is from its gravity field. However, perturbations, both gravitational and non-gravitational, can be significant especially when the particle is further away from the asteroid (Scheeres 2016). The two most significant sources of perturbations are from the Sun and we will discuss them briefly in this section.

4.4.1 SOLAR THIRD-BODY EFFECT

We consider a simple two-body problem, wherein the asteroid has a circular, heliocentric orbit in the ecliptic plane. This is a reasonable approximation, as mentioned earlier in Section 4.1, since several asteroids have been observed to have circular orbits around the Sun with near-zero inclinations. The gravitational effect of the Sun on the motion of regolith (henceforth STBE) around the asteroid is not modeled through a three-body problem because the order of magnitude of the perturbing acceleration is extremely small relative to the gravitational acceleration of the asteroid (at least five orders of magnitude smaller in the vicinity of a sample asteroid at 1 AU from the Sun) and hence it is sufficient to model it as an external perturbing acceleration.

The absolute gravity acceleration, due to the Sun, experienced by a particle (of mass negligible compared to that of the Sun) in orbital motion around the asteroid is given as (Scheeres 2016):

$$\vec{a}_{abs,p} = -\frac{\mu_S}{|\vec{r} - \vec{d}|^3}(\vec{r} - \vec{d}) \quad (4.39)$$

where μ_S is the gravitational parameter of the Sun; \vec{r} and \vec{d} are the position vectors of the orbiting particle and the Sun, respectively, from the asteroid's centre of mass, defined in the AIF. In Equation (4.39), the Sun is viewed to be orbiting the asteroid, instead of the other way around. This is just a change in perspective and is done to keep all distance vector definitions originating from the centre of mass of the asteroid (Scheeres 2016). The orientation of the position vectors is shown in Figure 4.5.

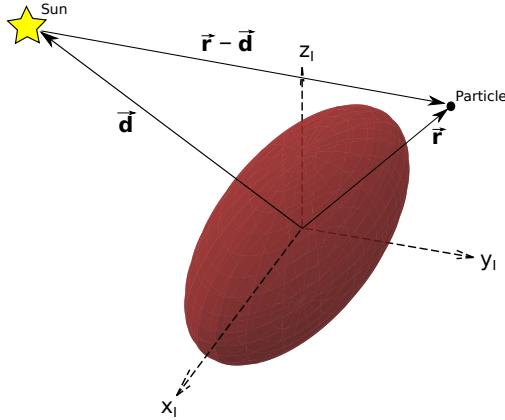


Figure 4.5: The orientation of position vectors of the Sun and the orbiting particle/regolith around the asteroid, in the AIF. Diagram is not to scale and the rotation state of the asteroid is such that the ARF and AIF are coinciding.

Now the absolute gravity acceleration, due to the Sun, experienced by the asteroid is given as follows (Scheeres 2016):

$$\vec{a}_{abs,a} = +\frac{\mu_S}{|\vec{d}|^3}(\vec{d}) \quad (4.40)$$

where the definition of all variables is the same as that for Equation (4.39). Thus, the relative perturbing acceleration acting on the particle due to the STBE is the difference between the absolute

accelerations experienced by the particle and the asteroid (Scheeres 2016):

$$\vec{a}_{STBE} = -\mu_S \left[\frac{(\vec{r} - \vec{d})}{|\vec{r} - \vec{d}|^3} + \frac{\vec{d}}{|\vec{d}|^3} \right] \quad (4.41)$$

The perturbing acceleration in Equation (4.41) can be re-written in the form of a potential as follows (Scheeres 2016):

$$\mathcal{R}_{STBE} = \mu_S \left[\frac{1}{|\vec{r} - \vec{d}|} - \frac{\vec{d} \cdot \vec{r}}{|\vec{d}|^3} \right] \quad (4.42)$$

$$\vec{a}_{STBE} = \frac{\delta \mathcal{R}_{STBE}}{\delta \vec{r}} \quad (4.43)$$

In Equation (4.41), we can directly substitute the position vectors as defined in the ARF such that the acceleration term obtained is also in the ARF. This is possible because the position magnitude terms would remain the same in either of the reference frames. Also, the rotation matrix ϕ_I^B multiplied either outside the square bracket or inside, in Equation (4.41), with the two numerator terms would ultimately give the same result.

$$\ddot{\vec{d}} = -\frac{\mu_S}{|\vec{d}|^3} (\vec{d}) \quad (4.44)$$

The apparent position of the Sun, relative to the asteroid, can be obtained in two ways. We could either numerically integrate the second-order differential equation for the standard two-body problem as stated in Equation (4.44) or solve, what is historically known as, *Kepler's problem*. We use the latter since the apparent position of the Sun can be obtained for any time value directly by using the Kepler's problem algorithm as stated by Chobotov 2002. Although the Kepler's problem algorithm is not completely analytical and uses a numerical iteration method to solve for the true anomaly, it is relatively easier to use within the simulator than employing a numerical integrator to propagate the position from an initial condition for every time value. The algorithm for solving Kepler's problem is not stated here for brevity; a detailed explanation for it is given by Chobotov 2002.

4.4.2 SOLAR RADIATION PRESSURE

With the heliocentric orbit of the asteroid, in addition to the STBE, is a non-gravitational source of perturbation acting on an asteroid-orbiting particle, called SRP. Momentum transfer takes place from the Solar photons that strike and recoil from the surface of the particle, which thus perturbs the orbital motion (Scheeres 2016).

We use a model for SRP that assumes that the particle always presents a constant area to the impinging Solar photons and the area is perpendicular to the Sun-line. The total momentum transfer is modeled as Solar irradiance and reflection and the acceleration due to SRP thus acts in a direction away from and along the Sun-line (Scheeres 2016). This acceleration is given as follows:

$$\vec{a}_{SRP} = -(1 + \rho) P_0 \cdot \frac{A}{M} \cdot \frac{(\vec{d} - \vec{r})}{|\vec{d} - \vec{r}|^3} \quad (4.45)$$

where ρ is the albedo of the particle; P_0 is a Solar constant whose value is 1.0×10^{17} kg m/s²; A/M is the area-to-mass ratio of the particle and area refers to the cross-sectional area of the particle on which the Solar photons are striking; \vec{d} is the distance vector from the asteroid to the Sun and \vec{r} is the distance vector from the asteroid to the orbiting particle (Scheeres 2016). The SRP perturbing

acceleration can be re-stated in terms of a perturbing potential as follows:

$$\mathcal{R}_{SRP} = -(1 + \rho) P_0 \cdot \frac{A}{M} \left[\frac{1}{|\vec{d} - \vec{r}|} \right] \quad (4.46)$$

$$\vec{a}_{SRP} = \frac{\delta \mathcal{R}_{SRP}}{\delta \vec{r}} \quad (4.47)$$

4.5 PERTURBED TWO-BODY PROBLEM

We have discussed the gravity potential and the perturbations model to be used for our system, and now we will present the Equations Of Motion (EOM) that govern the motion of the lofted particle. Note that the CDE potential model is defined for a body-fixed frame inherently which means the accelerations that we get out of it are directly defined for the ARF frame. The same applies for the STBE and SRP perturbing accelerations. This is why we define the equations of motion too in the ARF frame and transform the propagated state to the AIF frame post-simulation.

We use a Lagrangian approach to derive the EOM because once the Lagrangian is formed, it becomes relatively easy to re-write the EOM for a different frame of reference or a different set of coordinates by substituting the relevant transformation equations in the Lagrangian. It is formed as $L = T + \mathcal{U}$, where T is the specific kinetic energy and \mathcal{U} is the full potential of the system and they are given as follows (Scheeres 2016):

$$T = \frac{1}{2} \dot{\vec{r}}_I \cdot \dot{\vec{r}}_I \quad (4.48)$$

$$\mathcal{U} = U + \mathcal{R}_{STBE} + \mathcal{R}_{SRP} \quad (4.49)$$

where \vec{r}_I is the position vector of the particle from the asteroid to the orbiting particle and expressed in the AIF; U is the CDE gravity potential; \mathcal{R}_{STBE} and \mathcal{R}_{SRP} are the perturbing potentials for the Sun's third-body effect and radiation pressure respectively. The EOM is then obtained from the following (Scheeres 2016):

$$\frac{d}{dt} \left(\frac{\delta L}{\delta \dot{x}_i} \right) = \frac{\delta L}{\delta x_i} \quad (4.50)$$

where (x_i, \dot{x}_i) are the position and velocity vector components. We will now evaluate Equation (4.50) for our particular case by first re-stating the Lagrangian with vectors expressed in the ARF. With $(\vec{q}, \dot{\vec{q}})$ denoting the ARF position and velocity vectors respectively, the position vector in the AIF is related to the ARF as $\vec{r} = \phi_B^I \vec{q}$ and the velocity is related as $\dot{\vec{r}} = \phi_I^B \cdot (\dot{\vec{q}} + \vec{\omega} \times \vec{q})$. With these definitions, the Lagrangian is re-written as follows:

$$L = (\dot{\vec{q}} + \vec{\omega} \times \vec{q}) \cdot (\dot{\vec{q}} + \vec{\omega} \times \vec{q}) + \mathcal{U}(\vec{q}) \quad (4.51)$$

where the transformation matrix ϕ_B^I preserves ² the dot product, as it is orthogonal, and hence is excluded from the equation. The derivative on the right-hand side of Equation (4.50), directly in

²For two vectors \vec{u} and \vec{v} , and an orthogonal matrix Q , $\vec{u} \cdot \vec{v} = (Q \vec{u}) \cdot (Q \vec{v})$

vector form, is evaluated as follows:

$$\begin{aligned}
 \frac{\delta L}{\delta \vec{q}} &= \frac{1}{2} \left[2 \cdot (\dot{\vec{q}} + \vec{\omega} \times \vec{q}) \cdot (\vec{\omega} \times \vec{1} + \vec{q} \times \vec{0}) \right] + \frac{\delta \mathcal{U}}{\delta \vec{q}} \\
 &= \dot{\vec{q}} \cdot (\vec{\omega} \times \vec{1}) + (\vec{\omega} \times \vec{q}) \cdot (\vec{\omega} \times \vec{1}) + \frac{\delta \mathcal{U}}{\delta \vec{q}} \\
 &= (\dot{\vec{q}} \times \vec{\omega}) \cdot \vec{1} + \vec{1} \cdot ((\vec{\omega} \times \vec{q}) \times \vec{\omega}) + \frac{\delta \mathcal{U}}{\delta \vec{q}} \\
 &= -(\vec{\omega} \times \dot{\vec{q}}) - (\vec{\omega} \times (\vec{\omega} \times \vec{q})) + \frac{\delta \mathcal{U}}{\delta \vec{q}}
 \end{aligned} \tag{4.52}$$

where $\vec{1}$ and $\vec{0}$ are vectors of ones and zeros respectively and all other terms have been defined previously. The left hand side of Equation (4.50) is evaluated as follows:

$$\frac{\delta L}{\delta \dot{\vec{q}}} = \dot{\vec{q}} + (\vec{\omega} \times \vec{q}) \tag{4.53}$$

$$\frac{d}{dt} \left(\frac{\delta L}{\delta \dot{\vec{q}}} \right) = \ddot{\vec{q}} + \vec{\omega} \times \dot{\vec{q}} \tag{4.54}$$

Thus, by substituting Equations (4.52) and (4.54) into Equation (4.50), we get the final EOM as follows:

$$\begin{aligned}
 \ddot{\vec{q}} + 2 \cdot \vec{\omega} \times \dot{\vec{q}} + (\vec{\omega} \times (\vec{\omega} \times \vec{q})) &= \frac{\delta \mathcal{U}}{\delta \vec{q}} \\
 &= \frac{\delta U(\vec{q})}{\delta \vec{q}} + \frac{\delta \mathcal{R}_{STBE}}{\delta \vec{q}} + \frac{\delta \mathcal{R}_{SRP}}{\delta \vec{q}}
 \end{aligned} \tag{4.55}$$

$$\ddot{\vec{q}} + 2 \cdot \vec{\omega} \times \dot{\vec{q}} + (\vec{\omega} \times (\vec{\omega} \times \vec{q})) = \frac{\delta U(\vec{q})}{\delta \vec{q}} + \vec{a}_{STBE} + \vec{a}_{SRP} \tag{4.56}$$

In Equation (4.55), the first term denotes the acceleration due to gravity, and the second and the third term denote the perturbing accelerations. Equation (4.56) is thus the final EOM used in this thesis.

4.6 PARTICLE INITIAL CONDITIONS

The EOM established in the previous section are difficult to solve analytically and hence we make use of a numerical integrator to propagate the state vector of an orbiting particle in time. To do this, we need to establish robust methods for providing initial conditions. These initial conditions basically form the state vector with which the particle is lofted from the surface of an asteroid.

4.6.1 LAUNCH LOCATION

This section will present a method to efficiently calculate the launch location of a particle, in the form of a Cartesian position vector with respect to the centre of the asteroid. The formulation is such that the resulting position vector will always point to a location on the surface of the asteroid.

Consider the equation for the ellipsoid as follows:

$$\frac{x^2}{\alpha^2} + \frac{y^2}{\beta^2} + \frac{z^2}{\gamma^2} = 1 \tag{4.57}$$

where (x, y, z) is the coordinate of any point on the surface. An alternate way of writing Equation (4.57), in vector format, is given as follows:

$$\vec{r} \cdot E \cdot \vec{r} = 1 \quad (4.58)$$

$$E = \begin{bmatrix} 1/\alpha^2 & 0 & 0 \\ 0 & 1/\beta^2 & 0 \\ 0 & 0 & 1/\gamma^2 \end{bmatrix} \quad (4.59)$$

where \vec{r} is a general position vector expressed in the ARF and (α, β, γ) are the semi-axes of the triaxial ellipsoid model of the asteroid. Continuing with Equation (4.58):

$$\vec{r} \cdot \vec{r} = \frac{1}{E} \quad (4.60)$$

$$r^2 = \frac{1}{\hat{u} \cdot E \cdot \hat{u}} \quad (4.61)$$

where \hat{u} is a unit vector expressed in the ARF, pointing in the direction of the launch location from the asteroid's centre, with components (u_x, u_y, u_z) . The unit vector can be stated in terms of the latitude (δ) and longitude (λ) of the launch point as follows:

$$\hat{u} = \cos \delta \cos \lambda \hat{x} + \cos \delta \sin \lambda \hat{y} + \sin \delta \hat{z} \quad (4.62)$$

where $(\hat{x}, \hat{y}, \hat{z})$ are the basis vectors forming the ARF. Thus, Equation (4.61) can be written as:

$$r^2 = \frac{1}{(u_x^2/\alpha^2) + (u_y^2/\beta^2) + (u_z^2/\gamma^2)} \quad (4.63)$$

The final position vector to the launch location is given as follows:

$$\vec{r}_s = r \hat{u} \quad (4.64)$$

where r is obtained from Equation (4.63). Thus, just by specifying the latitude and longitude of any desired launch point, the position vector to it is obtained. This, thus, acts as the initial position state of the particle.

4.6.2 LAUNCH VELOCITY

Once we know the position of the particle, we need to provide an initial velocity to launch it off the surface of the asteroid. This has to do much more than just providing a velocity magnitude. The velocity magnitude is accompanied by a launch direction which is defined by two angles, namely Launch Azimuth (η) and Launch Declination (χ).

These angles can be understood as follows. Suppose you are standing on the surface of an asteroid and you throw a ball. Now the ball will be associated with a velocity vector and if we take the projection of this vector onto the local surface, then the angle between this projection and the local direction to the North pole of the asteroid is defined as the launch azimuth. The angle that the velocity vector itself makes with the local normal is then defined as the launch declination. By keeping a constant angle of declination and varying the launch azimuth from 0° to 360° , we can create a cone of particles, thus replicating how impact ejecta would launch out. An example of this is given in Figure 4.6.

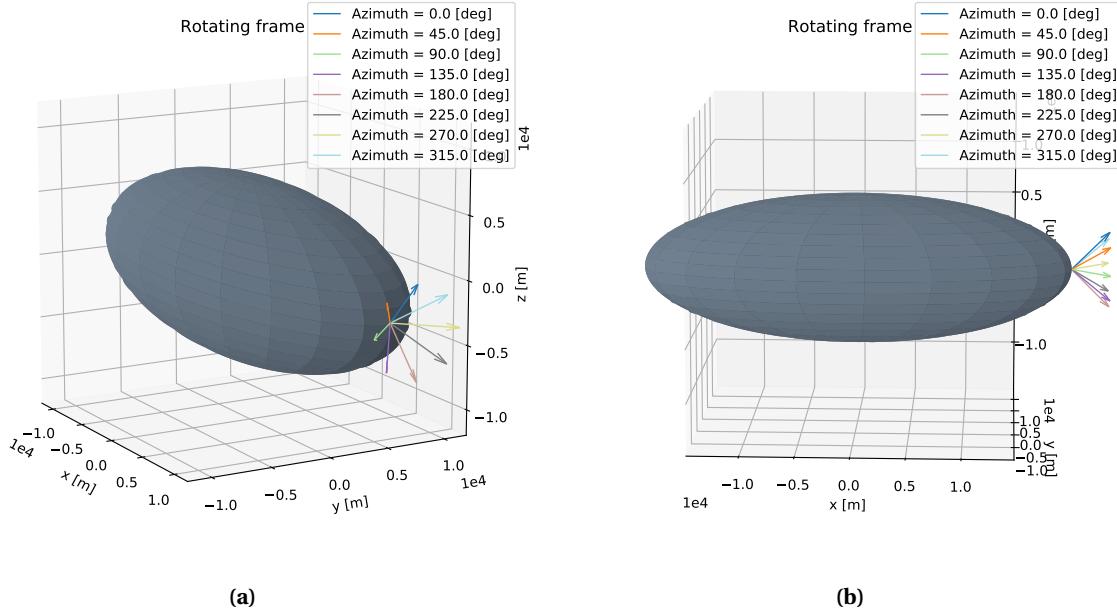


Figure 4.6: Various velocity vectors depicted at the longest edge of the ellipsoid shaped asteroid i.e. for a launch location longitude and latitude of 0° . Figure 4.6a depicts the front view of multiple velocity vectors spaced at 45° azimuth from each other and at a constant declination of 45° . All vectors are expressed in the ARF. Figure 4.6b depicts the lateral view of the same velocity vectors.

We will now discuss how the velocity vectors are formed in practice. Consider the equation for a triaxial ellipsoid given in Equation (4.57). Now for any given point (x, y, z) on the surface of the ellipsoid, the normal vector (\vec{n}) (and subsequently the unit normal vector (\hat{n})) to it is found by taking the gradient of Equation (4.57):

$$\vec{n} = \frac{2x}{\alpha^2} \hat{i} + \frac{2y}{\beta^2} \hat{j} + \frac{2z}{\gamma^2} \hat{k} \quad (4.65)$$

$$\hat{n} = \frac{\vec{n}}{|\vec{n}|} \quad (4.66)$$

where both \vec{n} and \hat{n} are expressed in the ARF, whose basis vectors are denoted here as $(\hat{i}, \hat{j}, \hat{k})$. Note that since the body in question is not a sphere, the normal vector at the launch location and the radial unit vector \hat{R} (from the centre of the ellipsoid to the launch location) may not always coincide. Now that we know the unit normal vector to the launch location, we need to find the local North direction. This is done by performing the following calculations successively:

$$\hat{R} = \frac{\vec{r}_s}{|\vec{r}_s|} \quad (4.67)$$

$$\hat{T} = \frac{\hat{R} \times \hat{k}}{|\hat{R} \times \hat{k}|} \quad (4.68)$$

$$\hat{x} = \frac{\hat{T} \times \hat{n}}{|\hat{T} \times \hat{n}|} \quad (4.69)$$

$$(4.70)$$

where \vec{r}_s is the position vector to the launch location from the centre of the asteroid; \hat{T} is the tangential unit vector (tangential to the local surface at the local point); \hat{x} is the unit vector that is

pointing towards the local North. This formulation works for any point on the surface of the asteroid to give the local North's direction except for the two poles and as such an alternative definition for them has to be used which will be defined later. Note that all the unit vectors discussed so far are expressed in the ARF.

\hat{x} is also viewed as the x-axis basis vector for a frame of reference fixed and centered at the launch location. This frame of reference is termed as the Surface Frame (SF). Note that the SF is not used to define the orbital motion of the regolith and hence it is not one of the standard reference frames. This is why it was not mentioned in Section 4.2 and is only defined here since its use is only for obtaining the velocity vector. The z-axis of the SF is defined in the direction of the unit normal vector at the launch location and the y-axis is obtained by following the right-hand rule to form the orthogonal frame. The SF is shown in Figure 4.7.

Note that the basis vectors of the SF are expressed in the ARF and thus their components are defined as follows:

$$\hat{x} = x_1 \hat{i} + x_2 \hat{j} + x_3 \hat{k} \quad (4.71)$$

$$\hat{y} = y_1 \hat{i} + y_2 \hat{j} + y_3 \hat{k} \quad (4.72)$$

$$\hat{z} = z_1 \hat{i} + z_2 \hat{j} + z_3 \hat{k} \quad (4.73)$$

The velocity vector components in the ARF are obtained as follows:

$$v_x = V [x_1 \cos \eta \sin \chi + y_1 \sin \eta \sin \chi + z_1 \cos \chi] \quad (4.74)$$

$$v_y = V [x_2 \cos \eta \sin \chi + y_2 \sin \eta \sin \chi + z_2 \cos \chi] \quad (4.75)$$

$$v_z = V [x_3 \cos \eta \sin \chi + y_3 \sin \eta \sin \chi + z_3 \cos \chi] \quad (4.76)$$

where V is the magnitude of the velocity chosen manually at the start of the simulation. Thus in this manner, and for the same velocity magnitude, we can simulate particles being launched in different directions from the same launch location, which in turn helps us understand the role of launch direction in the final fate of ejecta.

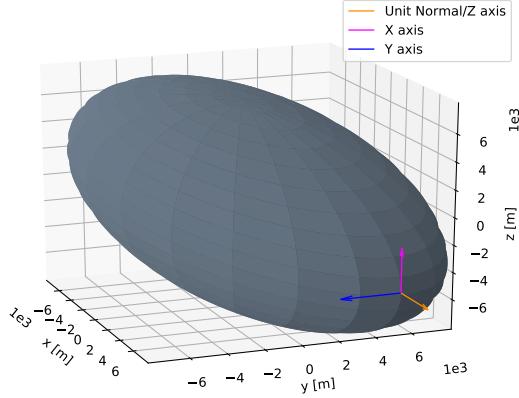


Figure 4.7: An example of the SF at the launch location of latitude and longitude 0° i.e. the longest edge of the asteroid. The x-axis points to the local North; z-axis is in the direction of the unit normal vector at the launch location; and finally the y-axis completes right-hand rule orthogonal frame. Two more generalized examples of the SF are illustrated in Figures B.2 and B.3.

As said earlier, the x-basis vector for the SF can not be defined by the method described previously for launch locations on the two poles of the asteroid. Hence we use two alternate definitions

for the x-basis vector in those scenarios; however all other computations remain the same as described before. If the launch location is the North pole, then the x-basis vector of the SF is defined as $\hat{\mathbf{x}} = [1, 0, 0]$ and for South pole it is defined as $\hat{\mathbf{x}} = [-1, 0, 0]$.

4.7 GUARANTEED ESCAPE SPEED

Up until now we discussed the dynamics involved with the particle motion around an asteroid and we also devised methods to launch regolith off the surface of the asteroid. It was mentioned earlier that the thesis has employed a full numerical simulation approach to the problem at hand, but while doing that we did attempt at finding a new analytical method to determine a Non-Conservative Guaranteed Escape Speed (NCGES); as opposed to a Conservative Guaranteed Escape Speed (CGES) method discussed by Scheeres [2016](#).

The CGES method is obtained by making use of the maximum gravity potential, between the actual potential of the irregular body and an equivalent point mass potential, at the location of the launch of the particle. For a body with non-uniform gravity field, if the launch speed is above the CGES, then the particle will definitely escape the asteroid and can not have any other final fate; however for launch velocities below the CGES, the particle can encounter any of the final fates, including escape. Thus in the case of a non-uniform gravity field, the CGES can not be used for a definitive predetermination of all launch velocities that will lead to an escape situation. The CGES method works best for uniform gravity field models, as we will see later in Chapter 5. In that case, it acts as a definitive demarcation between the final fates escape ($V_{\text{launch}} > \text{CGES}$) and re-impact ($V_{\text{launch}} < \text{CGES}$). The theory behind the CGES method is explained in Section 4.7.1.

The NCGES method on the other hand uses the property of conservation of the Jacobi integral and assumes a parabolic escape trajectory for the particle; the combination of these two is then used to derive a method which is completely different from the CGES. The theory behind NCGES is explained in more detail in Section 4.7.2. In both the CGES and the NCGES methods, the non-uniform gravity field of the asteroid is taken into account, however, no other external perturbations are included in their formulation.

The reason for working towards a NCGES algorithm is to account for the inadequacy of the CGES. As we will see later in Chapter 5, the CGES method fails to account for certain escape cases. Most of these cases involve particles completing one or more orbital revolutions around the asteroid before finally embarking on an escape trajectory. These are the cases that we hope to account for through the NCGES algorithm, in addition to the ones that are already accounted for through the CGES method.

4.7.1 CONSERVATIVE GUARANTEED ESCAPE SPEED

We will first understand how the CGES is obtained since it will help us in understanding the derivation for the NCGES later on. The inertial velocity of a particle which is resting on an asteroid's surface is given as $\vec{v}_I = \vec{\omega} \times \vec{r}$, where \vec{r} is the location to the particle from the centre of the asteroid. The resulting vector is pointing outwards from the asteroid if the launch location is on the leading edge of the asteroid and inwards if the launch location is on the trailing edge of the asteroid (Scheeres [2016](#)). This is depicted in Figure 4.8.

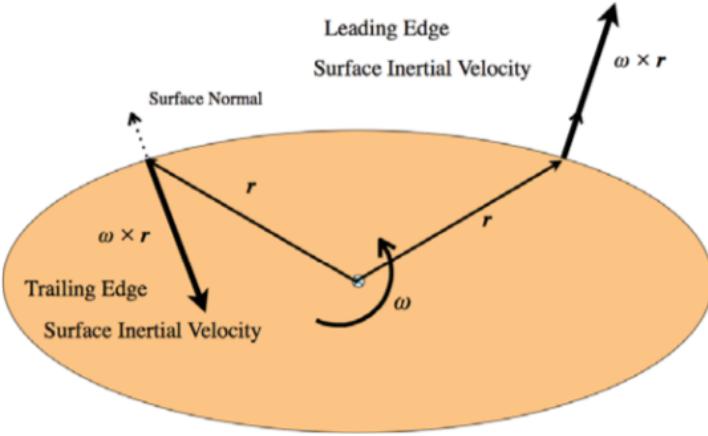


Figure 4.8: Particle inertial velocity at the surface for different launch locations (Scheeres 2016).

The idea now is to provide an additional launch speed, for instance in the normal direction, that would result in an escape scenario for the particle. The inertial velocity can then be expressed as follows (Scheeres 2016):

$$\vec{v}_I = v_e \hat{n} + \vec{\omega} \times \vec{r} \quad (4.77)$$

where $v_e \hat{n}$ is the velocity, expressed in the ARF, with which a particle is launched in the normal direction such that the particle is on an escape trajectory. All other terms have been defined previously. The conservative approach is to equate \vec{v}_I to $\sqrt{2U_{max}}$ where $U_{max} = \max[U(\vec{r}), \mu/\vec{r}]$ (Scheeres 2016). Equation (4.77) can then be re-written as follows:

$$\sqrt{2U_{max}} = v_e \hat{n} + \vec{\omega} \times \vec{r} \quad (4.78)$$

$$2U_{max} = v_e^2 + (\vec{\omega} \times \vec{r})^2 + 2v_e(\hat{n} \cdot (\vec{\omega} \times \vec{r})) \quad (4.79)$$

Equation (4.79) is a quadratic equation in v_e and can be solved as follows:

$$v_e = \frac{-2(\hat{n} \cdot (\vec{\omega} \times \vec{r})) \pm \sqrt{4(\hat{n} \cdot (\vec{\omega} \times \vec{r}))^2 + 4(2U_{max}) - 4(\vec{\omega} \times \vec{r})^2}}{2} \quad (4.80)$$

$$v_e = -(\hat{n} \cdot (\vec{\omega} \times \vec{r})) \pm \sqrt{(\hat{n} \cdot (\vec{\omega} \times \vec{r}))^2 + 2U_{max} - (\vec{\omega} \times \vec{r})^2} \quad (4.81)$$

Since speed can not be negative, the formula is re-written as:

$$v_e = -(\hat{n} \cdot (\vec{\omega} \times \vec{r})) + \sqrt{(\hat{n} \cdot (\vec{\omega} \times \vec{r}))^2 + 2U_{max} - (\vec{\omega} \times \vec{r})^2} \quad (4.82)$$

Equation (4.82) thus gives the CGES, expressed in the ARF, for a particle launched in the normal direction. The equation is equally applicable for a particle launched in any general direction as well by substituting \hat{n} with the unit vector for the direction in which the launch takes place. Thus if a particle is launched with a velocity above or equal to the one mentioned in Equation (4.82), then it would result in a guaranteed escape situation.

4.7.2 NON-CONSERVATIVE GUARANTEED ESCAPE SPEED

For the NCGES, we will not use $\vec{v}_I = \sqrt{2U_{max}}$ to get a guaranteed escape launch speed. We will derive an alternate relation for \vec{v}_I first and then later on substitute it back into Equation (4.82) in place of $2U_{max}$ to get the NCGES.

Consider the EOM mentioned in Equation (4.56), with the exception that we only consider the gravity potential and remove all external perturbations on the left-hand side of the equation. These equations do not have an explicit dependence on time which means that the Jacobian for the system exists and is conserved (Scheeres 2016). The Jacobian, expressed in the ARF, is given as follows (Scheeres 2016):

$$J = \frac{1}{2} v_B^2 - \frac{1}{2} \omega^2 (x^2 + y^2) - U(\vec{r}) \quad (4.83)$$

where v_B is the velocity of the particle in the ARF; ω is the magnitude of the angular velocity of the rotating asteroid and is along the z-axis of the ARF; (x, y) are the x-y coordinates of the orbiting particle; $U(\vec{r})$ is the small-body gravity potential, which in our case is the CDE potential model. From transport theorem, we use the relation between an inertial velocity and the rotating frame velocity:

$$\vec{v}_B = \vec{v}_I - (\vec{\omega} \times \vec{r}) \quad (4.84)$$

$$\begin{aligned} v_B^2 &= v_I^2 + (\vec{\omega} \times \vec{r})^2 - 2 \vec{v}_I \cdot (\vec{\omega} \times \vec{r}) \\ &= v_I^2 + (\vec{\omega} \times \vec{r})^2 - 2 \vec{\omega} \cdot (\vec{r} \times \vec{v}_I) \\ &= v_I^2 + (\vec{\omega} \times \vec{r})^2 - 2 \vec{\omega} \cdot \vec{H} \end{aligned} \quad (4.85)$$

where \vec{H} is the angular momentum of the orbiting particle. We substitute Equation (4.85) back into Equation (4.83) to get the following relation:

$$J = \frac{1}{2} v_I^2 - \vec{\omega} \cdot \vec{H} - U(\vec{r}) \quad (4.86)$$

Considering that after launch the particle is on a parabolic trajectory such that it just escapes then in such a case the energy of the particle would be 0 which reduces the Jacobian to $J_\infty = -\vec{\omega} \cdot \vec{H}_\infty$. Thus, J_∞ is the Jacobi integral's value when the particle is just outside the sphere of influence of the asteroid. Now at the instance of the launch, the Jacobian is expressed as follows:

$$J_0 = \frac{1}{2} v_{I_0}^2 - \vec{\omega} \cdot \vec{H}_0 - U(\vec{r}_0) \quad (4.87)$$

Since the Jacobian is conserved, $J_0 = J_\infty$, which means we can write the following relation:

$$\frac{1}{2} v_{I_0}^2 - \vec{\omega} \cdot \vec{H}_0 - U(\vec{r}_0) = -\vec{\omega} \cdot \vec{H}_\infty \quad (4.88)$$

For parabolic trajectories, the angular momentum magnitude is expressed in terms of the semi-latus rectum and the gravitational parameter of the central body as $H = \sqrt{\mu 2q}$ where q is the periapsis distance (Schaub et al. 2003). Also the angle between the angular momentum vector and the asteroid's rotation vector is the orbit inclination (i) and with these new definitions, Equation (4.88) is re-written as follows:

$$\frac{1}{2} v_{I_0}^2 - \vec{\omega} \cdot \vec{H}_0 - U(\vec{r}_0) = -\omega \cos(i) \sqrt{2\mu q_\infty} \quad (4.89)$$

We will consider only equatorial orbits for now to simplify Equation (4.89) by making the cosine term equal to one. Hence, Equation (4.89) can be stated as simply:

$$\frac{1}{2} v_{I_0}^2 - \vec{\omega} \cdot \vec{H}_0 - U(\vec{r}_0) = -\omega \sqrt{2\mu q_\infty} \quad (4.90)$$

$$\frac{1}{2} v_{I_0}^2 - \vec{\omega} \cdot (\vec{r}_0 \times \vec{v}_{I_0}) - U(\vec{r}_0) = -\omega \sqrt{2\mu q_\infty} \quad (4.91)$$

Note we can write $H_0 = \vec{r}_0 \times \vec{v}_{I_0}$, where \vec{r}_0 is expressed in the ARF, since at time $t = 0$ both ARF and AIF are aligned which means that $\vec{r}_{I_0} = \vec{r}_{B_0}$. The intent now is to solve for v_{I_0} from Equation (4.91), which is the inertial velocity at launch that eventually leads to an escape scenario.

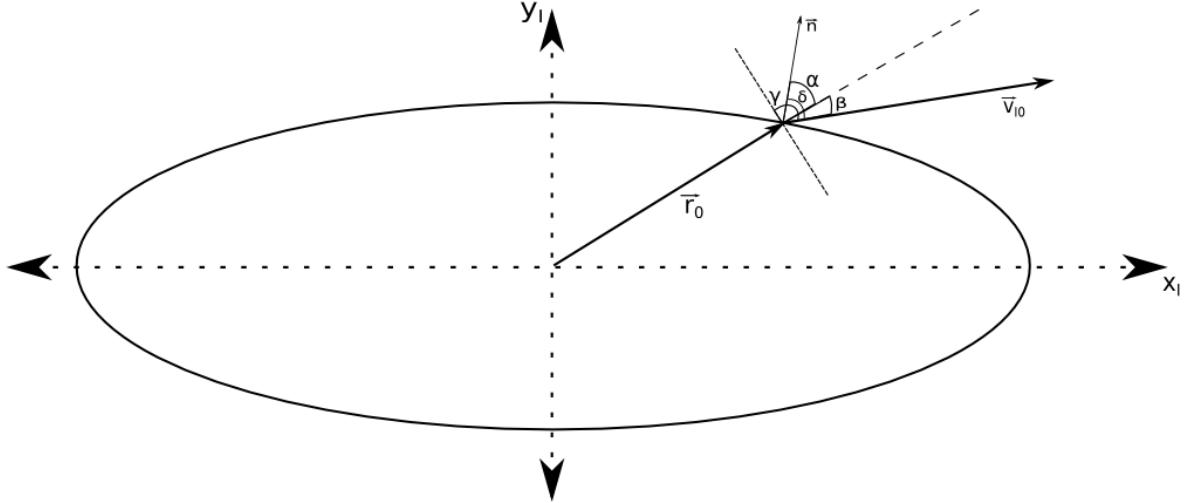


Figure 4.9: General angle definitions for velocity vector with position and normal vector at the launch location.

We need to evaluate the second term on the left-hand side of Equation (4.91) using the angle definitions given in Figure 4.9. In that, γ is the angle between the velocity vector and the plane perpendicular to the position vector; δ is the launch declination angle as explained in Section 4.6.2; α is the angle between the normal vector and the position vector direction and finally, β is the angle between the velocity vector and the position vector direction. Using these definitions, the cross product in Equation (4.91) is evaluated as follows:

$$\vec{\omega} \cdot (\vec{r}_0 \times \vec{v}_{I_0}) = \vec{\omega} \cdot (r_0 v_{I_0} \sin(\beta) \hat{h}) \quad (4.92)$$

$$= (r_0 v_{I_0} \sin(\delta - \alpha)) \vec{\omega} \cdot \hat{h} \quad (4.93)$$

where \hat{h} specifies the direction of the cross product, i.e. the unit vector of the angular momentum vector \vec{H}_0 . The angle definitions given in Figure 4.9 remain valid even for a launch location at the trailing edge of the asteroid and hence the angular definitions remain generalized. In addition to having an equatorial orbit, we can further simplify Equation (4.93) by keeping the launch site at the longest edge of the ellipsoid which means that we can make $\alpha = 0$. Thus Equation (4.91) can be re-written as:

$$\frac{1}{2} v_{I_0}^2 - (r_0 v_{I_0} \sin(\delta)) \vec{\omega} \cdot \hat{h} - U(\vec{r}_0) = -\omega \sqrt{2\mu q_\infty} \quad (4.94)$$

Equation (4.94) is a quadratic equation in v_{I_0} which is solved to provide the following solution:

$$v_{I_0} = (r_0 \sin \delta) \vec{\omega} \cdot \hat{h} \pm \sqrt{((r_0 \sin \delta) \vec{\omega} \cdot \hat{h})^2 + 2U - 2\omega \sqrt{2\mu q_\infty}} \quad (4.95)$$

Thus instead of using $v_{I_0} = \sqrt{2U_{max}}$ in Equation (4.79) and subsequently in Equation (4.82), as we did for the definition of the CGES, we use the formula in Equation (4.95) to define the NCGES. This leads to modifying Equation (4.82) as follows:

$$v_e = -(\hat{d} \cdot (\vec{\omega} \times \vec{r})) + \sqrt{(\hat{d} \cdot (\vec{\omega} \times \vec{r}))^2 + v_{I_0}^2 - (\vec{\omega} \times \vec{r})^2} \quad (4.96)$$

where v_{I_0} is defined in Equation (4.95) for the NCGES; and instead of using \hat{n} we use the unit vector \hat{d} , representing a general direction of launch and not just the normal direction.

Thus, the difference between the CGES and the NCGES is in the definition of the inertial guaranteed

escape velocity at launch. For the CGES, it is defined as $\sqrt{2U_{max}}$, and for NCGES it is defined by the expression in Equation (4.95). As mentioned in the introduction of Section 4.7, we work towards a NCGES method to also account for the escape cases where a particle completes one or more orbital revolutions around the asteroid before finally escaping. Due to a non-uniform gravity field, the trajectory of the particle is osculating and for an escape situation that does not occur immediately, the trajectory could osculate from a bounded circular or elliptical orbit to a parabolic or hyperbolic escape trajectory. Thus, with that in mind it should be noted that the value of q_∞ in Equation (4.95) is not determined from the initial launch position and velocity; rather, it is the value of the periapsis distance of the particle on a hypothetical parabolic trajectory, just outside the sphere of influence of the asteroid.

Note that the definition of the inertial guaranteed escape velocity at launch as expressed in Equation (4.95) is only valid when a particle is launched from a location such that the radial position vector to the launch location and the surface normal vector are aligned, and when the particle is launched (and stays) in an equatorial orbit around the asteroid. For any other case, the expression for v_{l_0} will have to be redefined since the cosine term in Equation (4.89) will not remain equal to one for a non-equatorial orbit and the angle α in Equation (4.93) will not remain zero when the surface normal and the launch location position vector are not aligned.

Part III

Numerical Simulation Results

5

GUARANTEED ESCAPE SPEED

In this chapter, we will discuss the results from the NCGES analytical method developed in Section 4.7. The algorithm's absolute performance and its viability is discussed and compared with that of the CGES algorithm.

5.1 SIMULATION SETUP

As mentioned in Section 4.7, the launch location for testing out the algorithm was chosen to be the longest edge of the ellipsoid shaped asteroid since it helped in simplifying the computations. In addition to this, the particles were launched in the equatorial plane such that the orbital inclination remained zero¹, which further simplified the computation.

We compare the results from our derivation of the NCGES with that of the conservative approach as defined by Scheeres 2016. To use the algorithm, the following launch conditions were used (these launch conditions are also depicted in Figure 5.1):

1. The launch location is at the longest edge of the ellipsoid.
2. The launch azimuth is equal to 270°.
3. The launch declinations are varied from 10° to 80°.
4. The launch velocity was kept fixed and chosen at random to be 6.0 m/s.

Note that these launch conditions result in equatorial orbits, just like how we want to test the NCGES algorithm. The value for the parameter q_∞ in Equation (4.95), which is the periapsis distance for a parabolic escape trajectory, was set manually for each simulated trajectory and remained constant during the entire duration of each simulation. Several values of q_∞ were used for testing and they were all taken as fractions of the largest dimension of the ellipsoid. This was done because we do not yet have a dedicated process to determine the value of q_∞ and hence several random fractions were used to gauge the output.

¹The orbital inclination remains zero valued after the launch in a non-uniform gravity field because the central body is homogeneous as well as symmetrical about the equator which means that there is equal attraction in the positive and negative z-axis directions which cancel each other out.

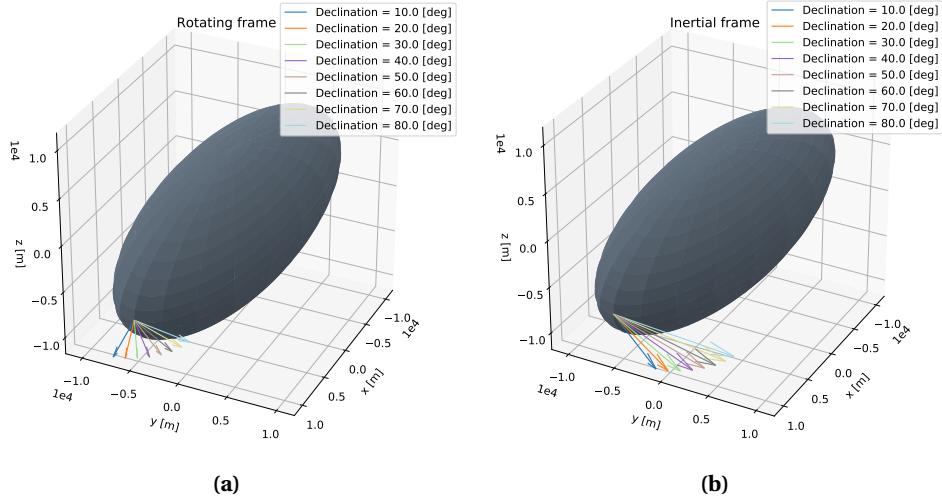


Figure 5.1: Launch vectors used for testing the NCGES algorithm. (a) Vectors expressed in the ARF and (b) Vectors expressed in the AIF.

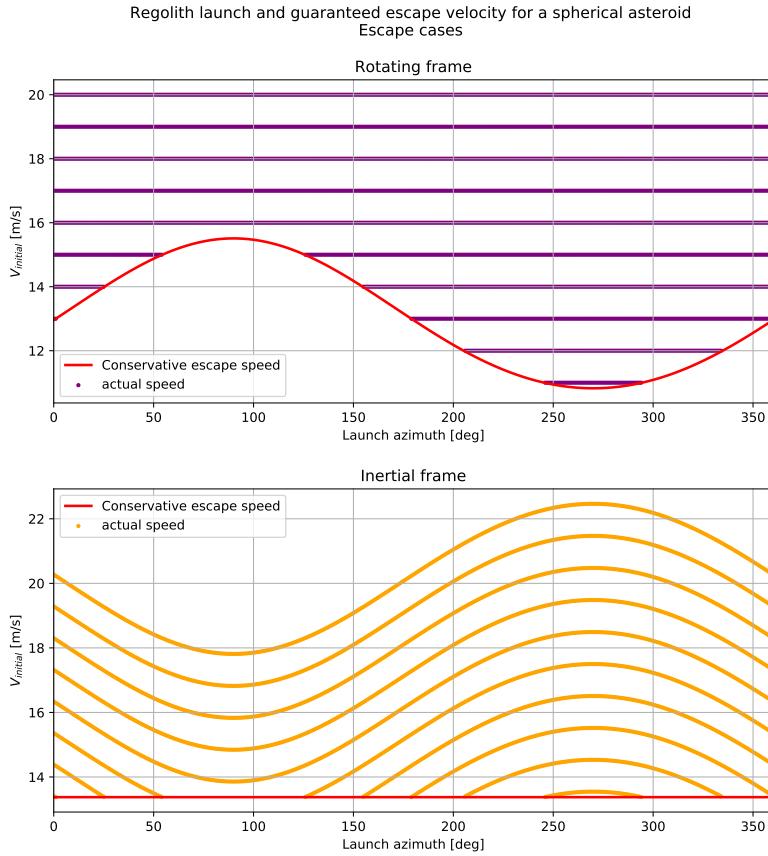


Figure 5.2: Escape velocities for varying launch azimuths and a constant launch declination of 45°. The CGES curve clearly separates all escape scenarios, as shown in both the AIF and the ARF. Note that for the launch azimuth of 270°, the CGES is lower than that for the launch azimuth of 90°. This is because in case of the former, the particle is launched in the direction of the rotation of the asteroid whereas in case of the latter, the particle is launched against the rotation of the asteroid.

5.2 CONSERVATIVE APPROACH WITH SPHERICAL ASTEROID

We will first look at the case of a homogeneous spherical asteroid whose radius is equal to 10 km. The simulation involved launching particles at a constant declination of 45° and for launch azimuth varying in the range $[0, 360]^\circ$. The launch velocities ranged from 1 - 20 m/s and the simulations did not include perturbations, gravity or otherwise. We used the CDE potential model but all three semi-axes were made equal to 10 km. In such a case, the CDE gravity potential model acts like a point mass potential model. This situation works for us since the latter is the actual gravity potential model for a point external to a homogeneous spherical body (MacMillan 1958).

The algorithm for the CGES works properly for the case of a homogeneous spherical asteroid, as shown in Figure 5.2, whose gravity potential is equivalent to that of a point mass. An escape occurs only if the particle was launched with a velocity which is equal to or above the CGES curve and not otherwise. Figure 5.3 shows how the curve even separates out all the re-impact cases. All launch velocities that are below the escape speed curve result only in re-impact and nothing else. Note that for this particular simulation we did not obtain any capture cases, but only escape and re-impact.

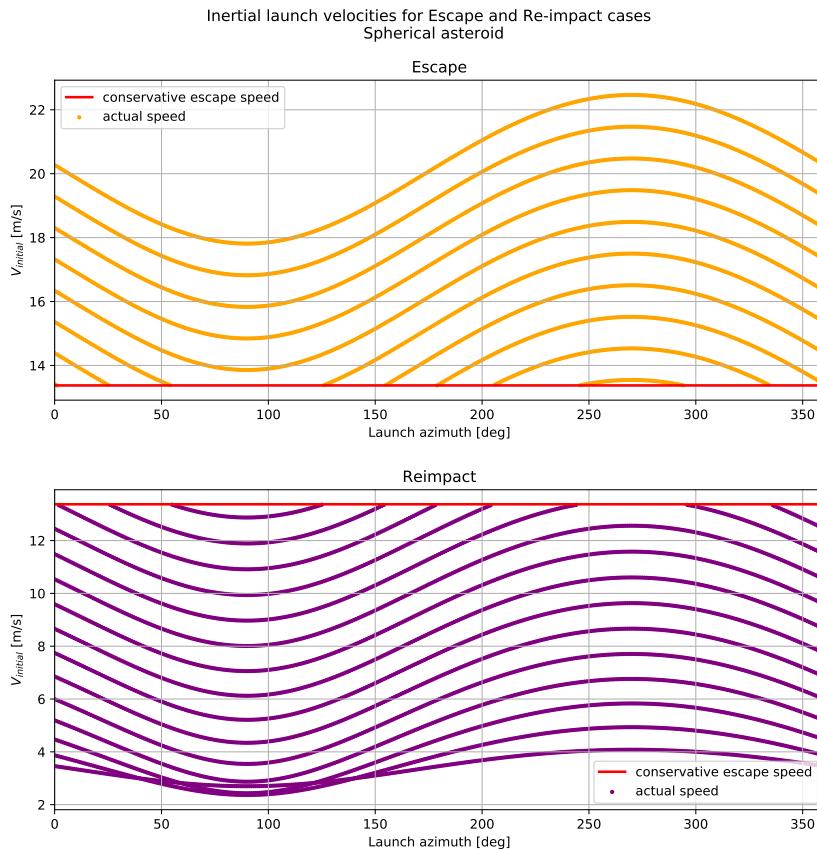


Figure 5.3: Escape and re-impact scenario velocities for varying launch azimuths and a constant launch declination of 45° . The CGES curve clearly separates all re-impact cases from the escape ones.

5.3 NON-CONSERVATIVE APPROACH WITH ELLIPSOIDAL ASTEROID

Now we shall look at the results for a CDE model, for which a particle was launched with the initial conditions enlisted earlier. Before we can analyze the result in Figure 5.4, it is important to discuss its structure. In the figure, the pink and the green region in the background depict the final fate

regime associated with the varying launch declination angles and the singular launch velocity of 6 m/s. The white spaces in between the pink and the green regions point out the cases which resulted in neither an escape nor a re-impact. These are also not capture cases as upon closer inspection, we saw that these cases have high altitude orbits with extremely high eccentricities such that the particles do not even complete a single revolution before the end of the simulation. Although the highlighted (pink or green) regions span the entire y-axis in Figure 5.4, they do not represent the final fate for any other velocity along that axis apart from the launch velocity of 6 m/s. Hence, those regions are also not fixed for any given range of declination angles.

Figure 5.4 shows the inadequacy of the CGES algorithm to predetermine escape scenarios based just on the initial conditions, as it fails to account for escape situations that occur at lower declination angles. In this particular situation, the particles are being lofted with a single value for the launch velocity i.e. 6 m/s and whenever this value is above the CGES curve (depicted by the black line in Figure 5.4), we see that the particle always escapes. This is how it should be and any result contrary to this means the simulator is at fault. The NCGES algorithm, however, does not function as expected. The algorithm was designed, hoping that it would also account for escape cases that the conservative approach was unable to. Figure 5.4 shows the NCGES curves for three different q_∞ values and although we get an idea on the performance from individual q_∞ values, the algorithm in general fails to identify escape scenarios since there are cases (at higher declination angles) where the launch velocity lies below the q_∞ curves but belongs to the escape regime.

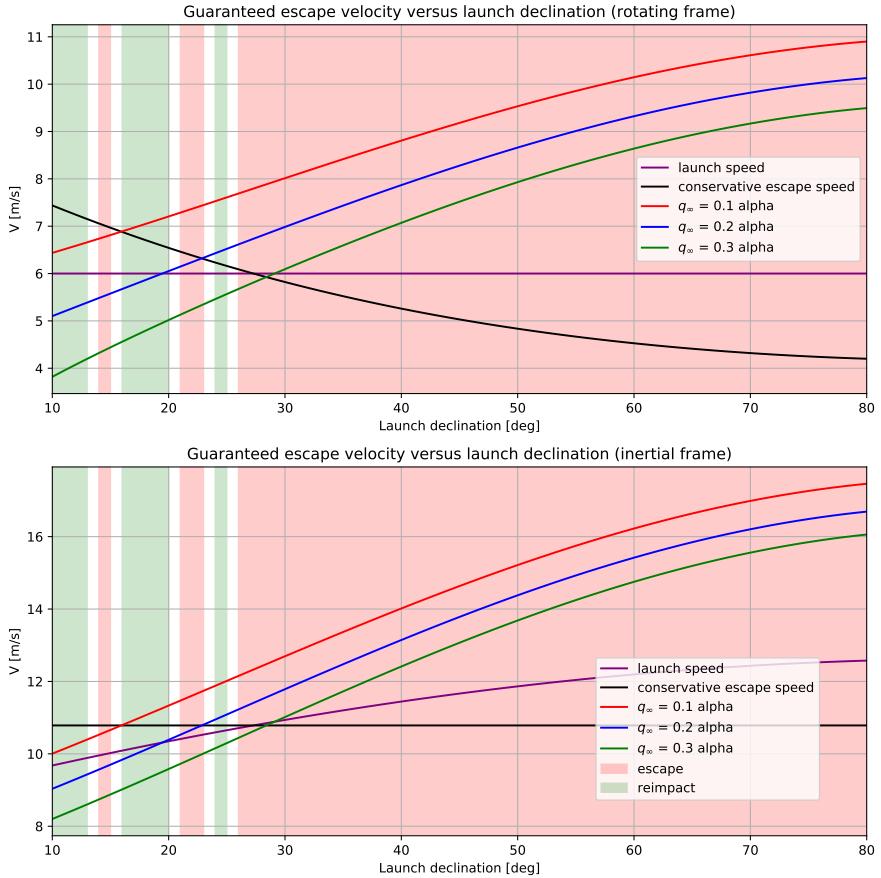


Figure 5.4: Escape and re-impact scenarios depicted for regolith launched with a single velocity and launch azimuth but multiple launch declination values. NCGES curve is shown for a CDE asteroid for three q_∞ values that are fractions of the largest semi-axes, α , of the ellipsoid. The CGES curve is also shown for comparison.

It is important to note that in Figure 5.4, the NCGES curves used only the "+" sign part of the formula in Equation (4.95) and not the "-" sign part since the latter always gave negative velocities for multiple sample values of q_∞ . We performed the simulation for the NCGES approach for the same launch azimuth and range of declination angles as before but velocities ranging from 1 to 16 m/s and $q_\infty = 0.3$ to see if the curve provides better escape estimates at other launch velocities. The result of this is shown in Figure 5.5. We see yet again the failure of the so called NCGES algorithm. We witness that even when the launch velocity is above the NCGES curve, we have re-impact scenarios. This clearly means that the algorithm is not even able to demarcate re-impact situations. At least with the CGES curve, we know that if the launch velocity is above the curve then the regolith can only escape and not have any other final fate.

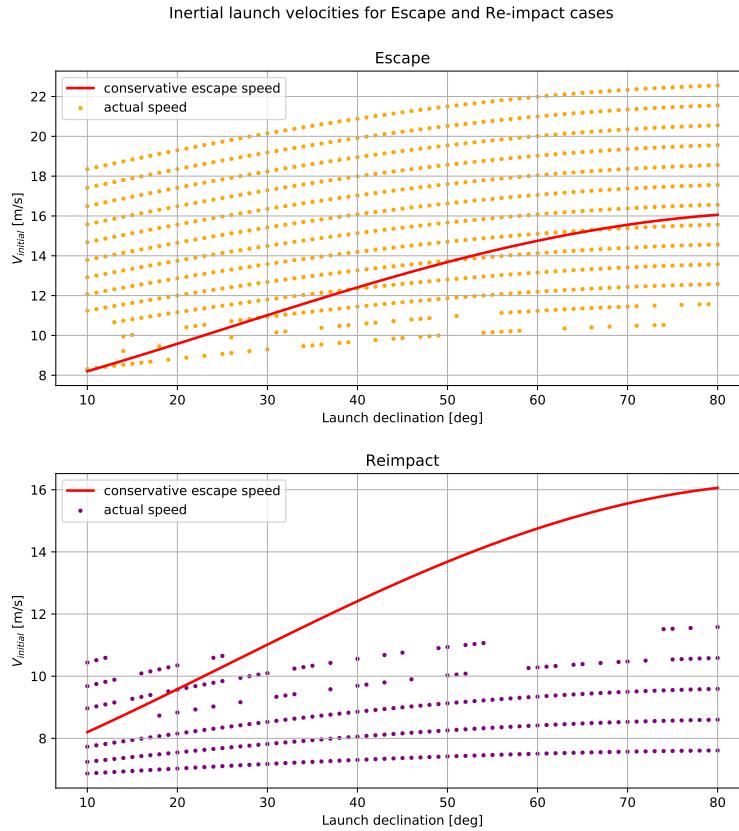


Figure 5.5: Escape and re-impact scenarios depicted for regolith launched from the longest edge of CDE with multiple velocities with launch azimuth = 270° and launch declination in the range of 10 to 80°. The NCGES curve is shown for $q_\infty = 0.3$.

On the other end of the spectrum, we can see that there are launch velocities below the NCGES curve where escape scenarios occur. This is another indication of the failure of the algorithm we designed.

Note that in Figure 5.5, the distribution of data points between re-impact and escape regimes is irregular within the mid-range launch velocities. It was observed that for the same launch velocity magnitude, non-uniform gravity field and a rapid rotation rate of the asteroid, the direction in which the particle is launched plays an important role in deciding the final fate of the regolith. Let us consider an example where the launch velocity is 5 m/s (expressed in the ARF) but the launch declinations are 40° and 41°. For the former, the final fate is re-impact whereas for the latter, the

final fate is escape. Upon comparing the x-y plane trajectory plot for the two cases, shown in Figure C.2, it is observed that the trajectories for them are very similar; however, after completing the one and only revolution around the asteroid, the particle launched with a declination angle of 41° is relatively further away from the asteroid. This means that the (rapidly) rotating asteroid did not trap the particle and cause it to re-impact, as it did for the case of launch declination 40° since it was closer to the asteroid, but rather assisted it such that it gained enough velocity to escape.

In another example, as shown in Figure 5.6, the launch declination is fixed at 10° but the launch velocities considered are 4 m/s and 5 m/s. In that, the former results in an escape situation while the latter results in re-impact. This time, the changes in the launch conditions are not subtle as they were in the previous example with just one degree of a change in the launch declination, but in fact are much more significant with a 1 m/s difference in launch velocities.

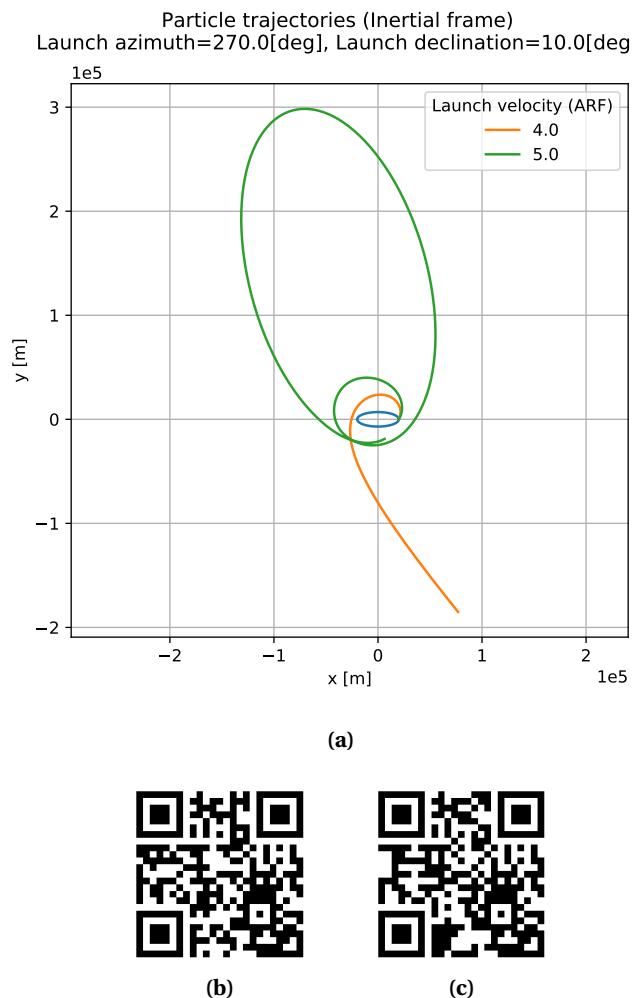


Figure 5.6: Particle trajectories and animation links for the same launch declination of 10° but for two different launch velocities, launched from the longest edge of a CDE asteroid. (a) shows the equatorial trajectories for the two velocities.

To view the trajectory animation for the 4 m/s case, scan the QR code in (b) or access the web-link

<https://youtu.be/u1ulpI04mbeE>, and for the 5 m/s case, scan the QR code in (c) or access the web-link
<https://youtu.be/S9A7oAmmD1g>.

For the case of 4 m/s, we see from the animation in Figure 5.6b that the particle is relatively closer to the asteroid after being lofted and gets pulled by the rapidly rotating asteroid and has its velocity increased enough to escape. But in the case of 5 m/s, we observe from the trajectory animation in

Figure 5.6c that after being lofted, the particle trajectory is relatively further away compared to the 4 m/s case. This results in a relatively weaker effect of the gravitational pull from the asteroid while it is rotating which causes the particle to stay in a bounded orbit before eventually re-impacting the surface.

Apart from the inability to completely differentiate between escape and re-impact cases, as we observed in Figure 5.5, another problem with the NCGES algorithm is that if we increase the value of q_∞ beyond certain extent, then we obtain a meaningless curve for the NCGES in the ARF. An example of this is depicted in Figure C.3 where $q_\infty = 0.7$. The curve as expressed in the ARF has no meaning since negative speeds are not valid. Thus for a relatively high q_∞ value, the output of NCGES provides invalid values of speeds, thereby pointing to the failure of the method on two accounts.

The NCGES method did not work as expected in identifying the escape cases that were undetected by the CGES method. In addition to this, we saw that the method produces a valid escape velocity curve only for a small range of q_∞ values. Although the NCGES method failed, the approach to derive it was correct and now we know that even if it sounds reasonable in theory, it fails completely in practice.

5.4 CONSERVATIVE APPROACH LIMITATIONS WITH ELLIPSOIDAL ASTEROID

The CGES approach did not account for a few escape scenarios when regolith was launched from the surface of a CDE-shaped asteroid and the reason for that is the combined effect of the shape/gravity field variations and a rapid rotation rate of the asteroid. For example, from the trajectory for the regolith launched at a declination angle of 15° in Figure 5.4, it was observed that the particle completes one revolution around the asteroid before embarking on a final hyperbolic trajectory. This is shown in Figure 5.7.

The animation for the trajectory in Figure 5.7 can be found at the web-link given in Figure 5.8. The animation clearly shows the rapid rotation rate of the asteroid which accelerates the particle as it approaches behind it and completes the one and only revolution around the asteroid; having its velocity increased enough to eventually attain a positive energy and escape the asteroid. Thus with the help of gravity perturbations and a fast rotating asteroid, the particle changes from an elliptical orbit to a hyperbolic trajectory leading to its escape. This behavior can not be easily captured just from the initial conditions, as we observed, by the CGES algorithm.

An important thing to note here is that the initial condition of the regolith was such that its osculating eccentricity was above 1.0 (but a negative total energy), meaning that at the moment of launch the regolith was on a hyperbolic trajectory but instantly evolves its orbit into an elliptical one. The reason for this is again the perturbations from a non-uniform gravity field and a rapidly rotating asteroid which keeps osculating the orbit. On the other hand, when we consider a spherical asteroid, the launched particle continues to propagate on the initial trajectory itself because in the absence of perturbations the initial orbital elements do not osculate. Thus for a spherical asteroid, a particle can only escape if the initial orbital elements are such that it is on a parabolic or hyperbolic trajectory and not otherwise due to a lack of external influences to osculate the orbit. This is why the CGES algorithm works for the spherical asteroid in predetermination of escape situations.

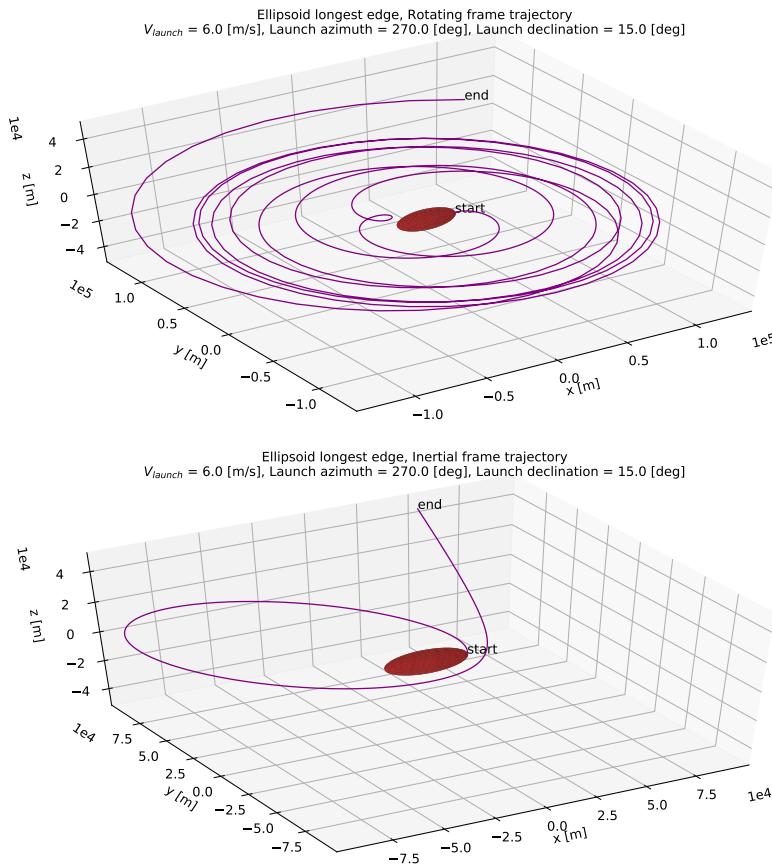


Figure 5.7: 3D trajectory in the ARF and the AIF for launch declination angle 15° from Figure 5.4.



Figure 5.8: Scan the QR code to view the 2D trajectory animation in the AIF for launch declination angle 15° from Figure 5.4. The video can also be accessed from the following web-link: https://youtu.be/51_CAYnjotk.

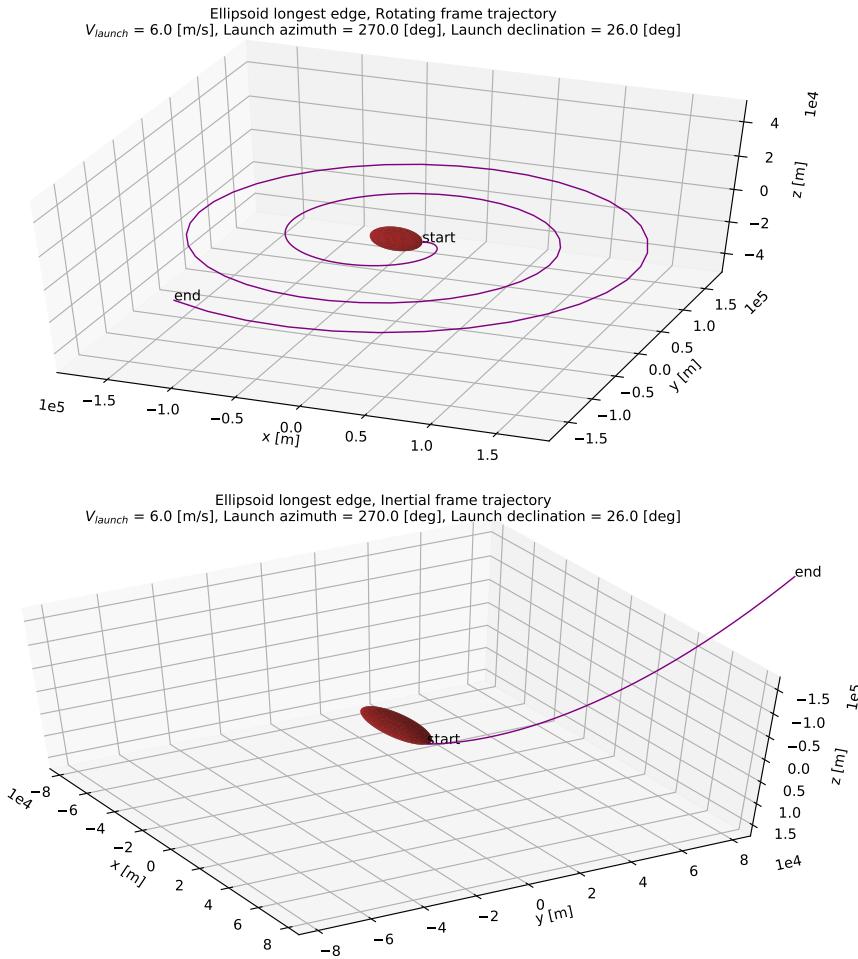


Figure 5.9: 3D trajectory expressed in the ARF and the AIF for a launch declination angle of 26° from the normal direction. The particle launch conditions are the same as that used for Figure 5.4.

The final aspect that will be discussed in regard to the CGES algorithm is its capacity to distinguish between particles that escape immediately and the ones that take one or more revolutions before escaping, when launched from an irregular body (Scheeres et al. 2002). However, we found out that this is only true if the particle is launched from the surface in the normal direction. In that, if the launch speed is lower than the CGES and the particle escapes, then it underwent multiple revolutions. This was observed for multiple particles launched from the longest edge of the CDE for launch velocities ranging from 1 to 16 m/s. However, this phenomenon is not observed when the launch direction is not in the normal direction. For example, from Figure 5.4, the launch declination angle of 26° results in a direct escape scenario, even though the launch velocity is below the CGES algorithm. The 3D trajectory for this case is shown in Figure 5.9.

Thus, it is imperative to understand that with just a non-uniform gravity field and a relatively fast rotating asteroid, the dynamics for orbiting regolith become intangible enough such that predetermination of orbital behavior and final fate of the regolith can not be explained by simple analytical methods. We attempted to explain the complex behavior by deriving a different guaranteed escape speed algorithm, the NCGES, however the method failed completely. We realize now that a numerical simulation method is a relatively better approach to understand the orbital dynamics of regolith lofted from an asteroid.

6

DYNAMICS WITHOUT SOLAR PERTURBATIONS

In this chapter, we will analyze the orbital motion of regolith in presence of a non-uniform gravity field, i.e. the CDE gravity model, but in the absence of all Solar perturbations. We try to understand the general behavior of regolith under such circumstances first so that we can later understand the effect of adding Solar perturbations.

6.1 SIMULATION SETUP

Since the aim is to understand the general characteristic behavior of regolith orbiting an elongated asteroid with a non-uniform gravity field, we chose the launch site to be on the longest edge of the asteroid, i.e. longitude and latitude both 0° . This launch site was chosen because it offers an easy and intuitive demarcation of particles being launched in or against the direction of asteroid's rotation. This demarcation is non-intuitive for any other general launch site, such as the one on the leading edge of the asteroid.

Multiple particles are launched from the surface, forming the shape of a cone. The launch declination angle was fixed to be a general value of 45° and the launch azimuth angle was varied from 0 to 359° , thereby launching regolith in every possible direction for a given declination value. The launch velocity was varied from 1 to 16 m/s. Thus, in total 5760 particles were launched and simulated, each with unique initial conditions.

6.2 FINAL FATE CHARACTERISTICS

The results presented in this section are a direct consequence of an attempt to understand any link between how a particle was launched and what their final outcome is. This, of course, was done in the absence of any external disturbance apart from the non-uniform gravity field so as to reduce the number of contributing factors for uncertainty. The initial intention of performing this part of the research was to discover the said link(s) through investigative methods and then later exploit it for space exploration activities.

6.2.1 GENERAL BEHAVIOR

We will begin by first looking at the different final outcomes and understand them as is by observing certain underlying parametric values. Figure 6.1 depicts the final fates acquired by particles for each of the launch velocities (expressed in the ARF). At the extreme ends of the launch velocity range, we can see that the final fate is absolute. If the launch velocity is extremely small then all particles just

re-impact the surface and if the velocity is very large, then the particles simply escape. For all other launch velocities in between, the distribution of final fates is mixed, with re-impact gaining more prominence over escape if the velocity is relatively lower and vice-versa.

We compare the final fate behavior in Figure 6.1 for an ellipsoidal asteroid, with the escape behavior for a spherical asteroid shown in Figure 5.2. For the former, it is observed that escape situations occur for launch velocities as low as 4 m/s while for the latter, they start occurring only at 11 m/s. The complete re-impact/escape histogram for the spherical asteroid is shown in Figure C.4. Clearly, the number of escape cases for the ellipsoidal asteroid is much higher than that for the spherical asteroid. There are two reasons for this, first being the relatively higher inertial speed of the particle at rest for the given launch site on the ellipsoidal asteroid, i.e. 6.623 m/s, compared to just 3.312 m/s for the spherical asteroid. The difference is because the longest axis of the ellipsoidal asteroid, from where the particles were lofted, is 20 km; whereas in case of the spherical asteroid, the particles were launched from an equatorial location and the radius of the sphere was 10 km. The second reason for the lower number of escaped particles in case of the spherical asteroid is its relatively higher gravity potential value at the launch site, i.e. $89.4403 \text{ m}^2/\text{s}^2$, compared to $58.155 \text{ m}^2/\text{s}^2$ for the ellipsoidal asteroid.

If we launch particles from an equatorial site on a spherical asteroid of radius 20 km for the same launch conditions as the ellipsoidal asteroid case, then the inertial speed of the particle at rest is the same for both asteroids but the gravity potential is $357.761 \text{ m}^2/\text{s}^2$ at the launch site. In such a situation, all particles re-impacted and there were no escape cases since the guaranteed escape speed was much higher and all launch velocities (1 to 16 m/s) were below it. The variation in the CGES curve with launch azimuth, for the spherical asteroid of radius 20 km, is shown in Figure C.5.

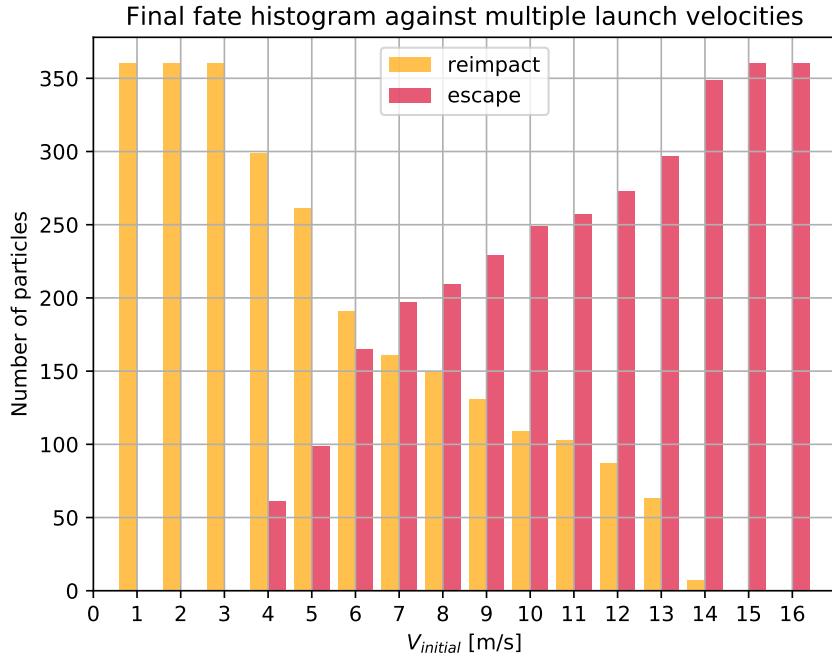


Figure 6.1: Re-impact/escape outcomes against launch velocities, which are expressed in the ARF. The values on the y-axis are representative of the number of particles that acquired each of the distinctive final fates for every launch velocity and are not to be confused with the launch azimuth values.

An important thing to notice is the lack of particles with temporary capture as their final fate

in Figure 6.1. Recall from Section 4.1 that each particle or regolith is simulated for a maximum of 270 days. So upon closer examination of the output databases, it was found that there are a few particles which did not re-impact nor escape at the end of the 270 days simulation. These particles were not temporarily captured by the asteroid, but instead, each particle was found to be on a single orbit with extremely large semi-major axis and eccentricity. Soon after launch these particles went extremely far away from the asteroid, but still weakly bounded to it through gravity, and never returned back in the vicinity of the asteroid within the total time set for the simulation. These are not capture orbits since the definition requires them to perform multiple orbital revolutions around the asteroid for hundreds of days. Moreover, these are the particles that eventually get whisked away from the asteroid's gravity pull because of Solar perturbations (observed when simulations were run with SRP and STBE).

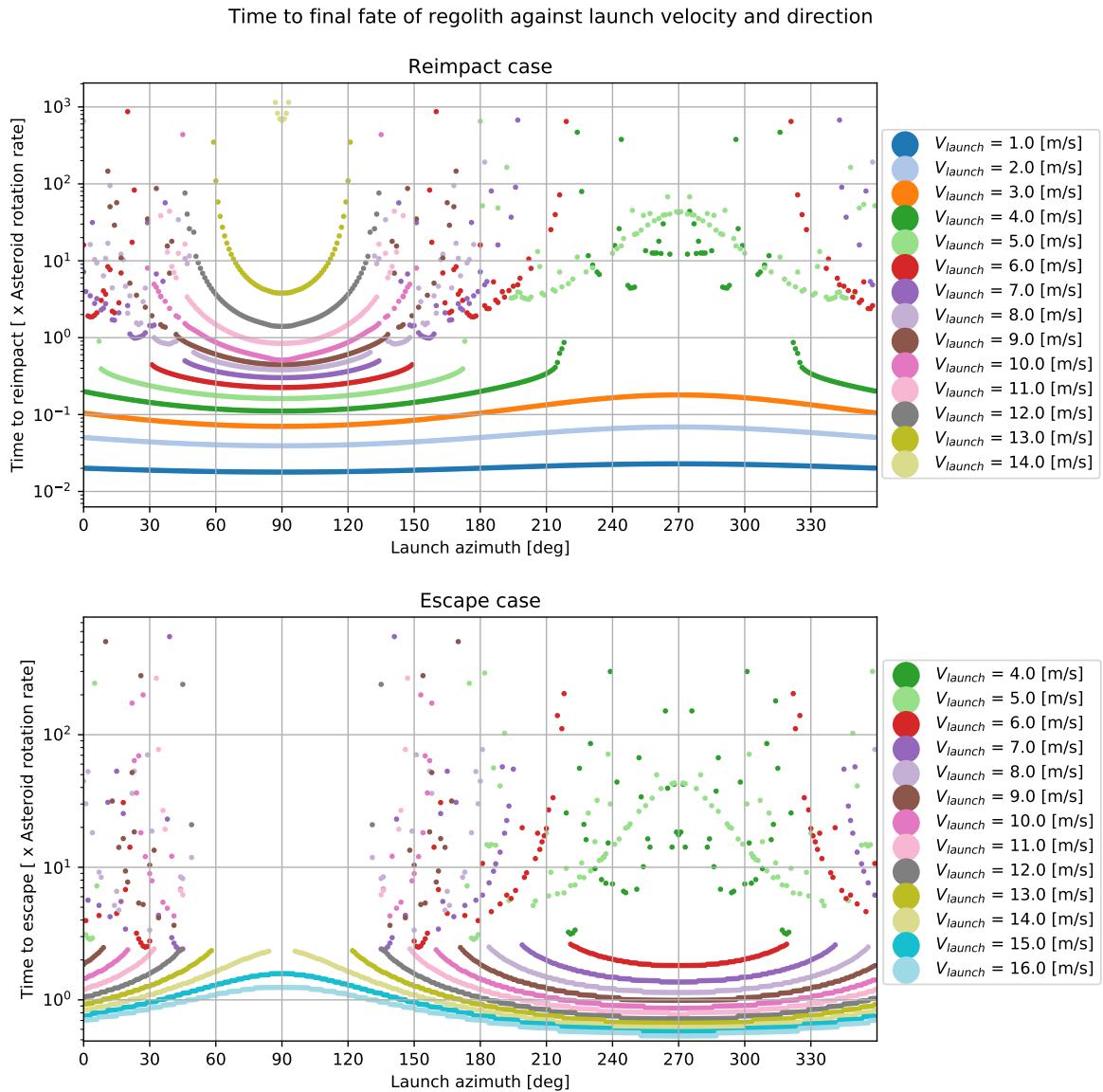


Figure 6.2: Total time taken by regolith to meet their respective final outcomes, as a function of launch azimuth and launch velocity. The time values are mentioned as a multiple of the asteroid rotation rate which is 5.27 hours.

Figure 6.2 shows the total time taken by regolith, launched with different launch conditions, to

meet their respective final fates. The plot shows that, for the same final fate, it's not just the launch velocity but the direction of launch as well that can have a significant impact on the total time it takes to meet a given fate. For a given launch velocity and say the escape case, we can see that if the particle is launched in the direction of the rotation (for example, azimuth = 270°), then it gets an extra boost from the asteroid's rotation and results in taking lesser time to escape. This is relative to being launched in a direction opposite to the asteroid's rotation (for example, azimuth = 90°), where the particle's inertial velocity reduces and hence takes more time to escape.

This situation with launch direction and time to final fate gets reversed when we consider the case of re-impact. Thus for the same velocity, when launched in the direction of asteroid's rotation, the particle tends to go further away from the asteroid because of an excess velocity provided by the rotation which then results in more time for the particle to come back and re-impact the surface. This directional behavior, for escape or re-impact, is seen for the majority of the cases, however, it can not be generalized as we do see some exceptions. These exceptions are observed only for the cases where the launch velocity results in both escape and re-impact situations. The most notable example for this is the escape scenarios for the launch velocity of 4 m/s. For the current example, escape occurs only when the launch azimuth is favorable, i.e. when the particles are launched in the rotational direction, in addition to a complex interaction with the fast rotating asteroid and irregular gravity field. So for the example of 4 m/s, escape occurs not because the velocity is inherently high enough but because of the interaction with the asteroid, which is why a chaotic behavior for it is observed in Figure 6.2.

We now look at the velocity extremes in Figure 6.1 which result either only in a re-impact or escape, and observe their orbital time in Figure 6.2. It is observed that the time to achieve the respective final fates is extremely small and mostly in fractions of the asteroid's rotation rate. This means that these particles do not spend enough time in orbit to get affected by the non-uniform gravity field or the asteroid's rotation and hence show no exceptional behavior as discussed before.

6.2.2 RE-IMPACT BEHAVIOR

This section will discuss certain characteristic behavior that is specific to the re-impact scenario. Figure 6.3 shows a map of re-impact locations on the surface of the asteroid for all launch velocities that led to this particular fate. The distinction due to launch azimuth is not plotted explicitly. The qualitative map in the plot is simply a 2D histogram plot with hexagonal shaped bins. These bins do not represent the number of particles within a given location but instead represent the initial launch velocities that led to a re-impact in that location. From the qualitative map, we can observe that particles launched with the lowermost velocities mostly re-impact very close to the launch site itself. The intermediate velocities seem to confine the re-impact locations within the -30° to 30° latitude range and lie mostly on the West of the launch site. And finally, the higher velocity range cause the particles to re-impact mostly on the upper latitudes in both the North and the South.

Thus the qualitative map gives an excellent description of how the re-impact locations change with increasing launch velocities. The map with the actual re-impact locations in Figure 6.3 shows that the bulk of all re-impacted particles lie around the two longest edges of the asteroid. Everywhere in between has a very scarce distribution of particles. A pattern in the re-impact location becomes apparent if one visualizes the launch azimuth angles, especially for the lower launch velocity range and is discussed shortly.

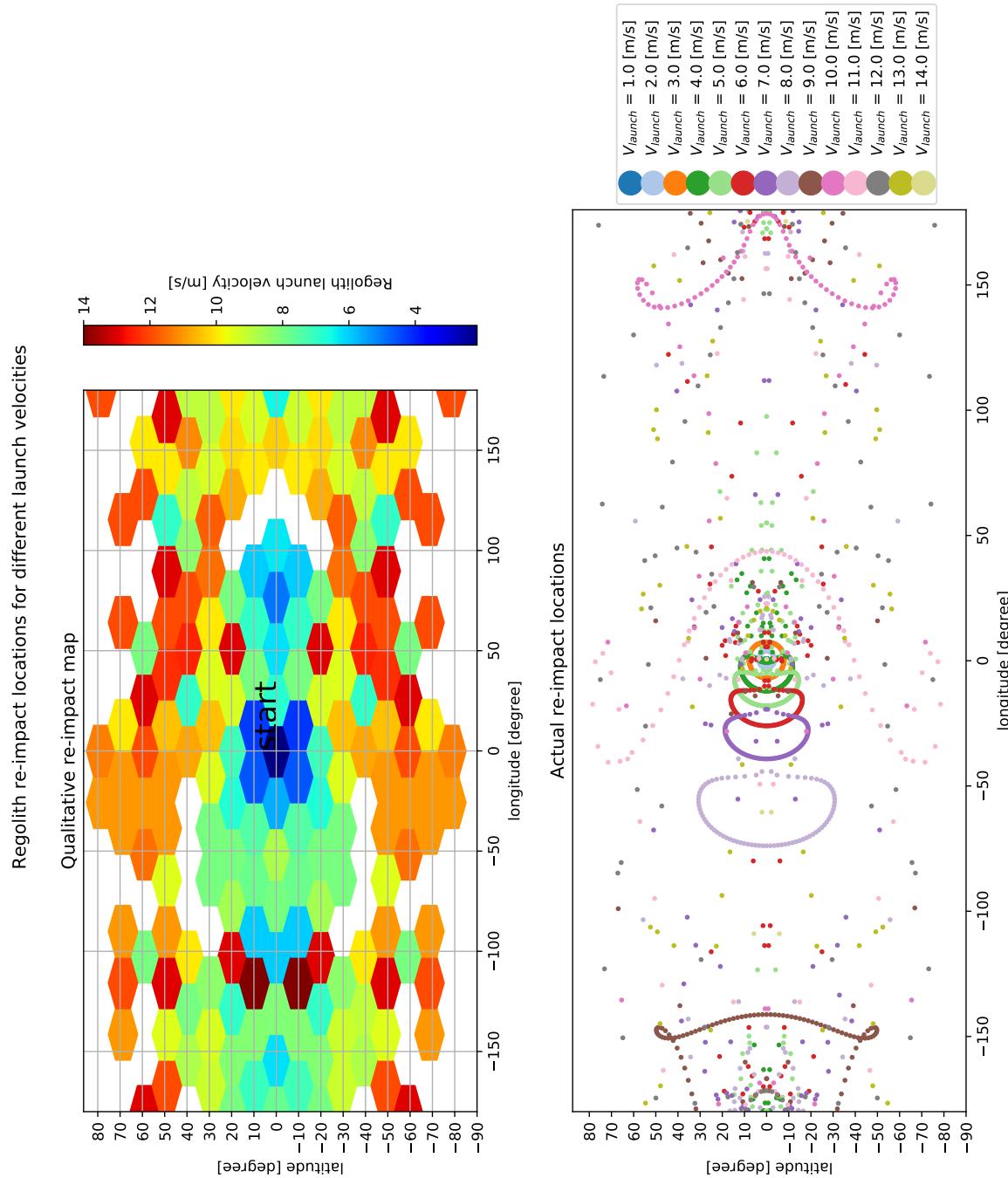


Figure 6.3: Regolith re-impact locations marked for all launch velocities that eventually lead to it. A qualitative map is shown as well that provides a general idea on re-impact locations and is color coded based on the regolith launch velocity.

We look a bit more in detail into the re-impact location map for three different velocities. These velocities are chosen such that, by studying them, we can estimate the behavior for the low, intermediate and high launch velocities. In addition to this, we also look at the effect of the launch direction on re-impact location and see if there is a link between them or if it is all chaotic.

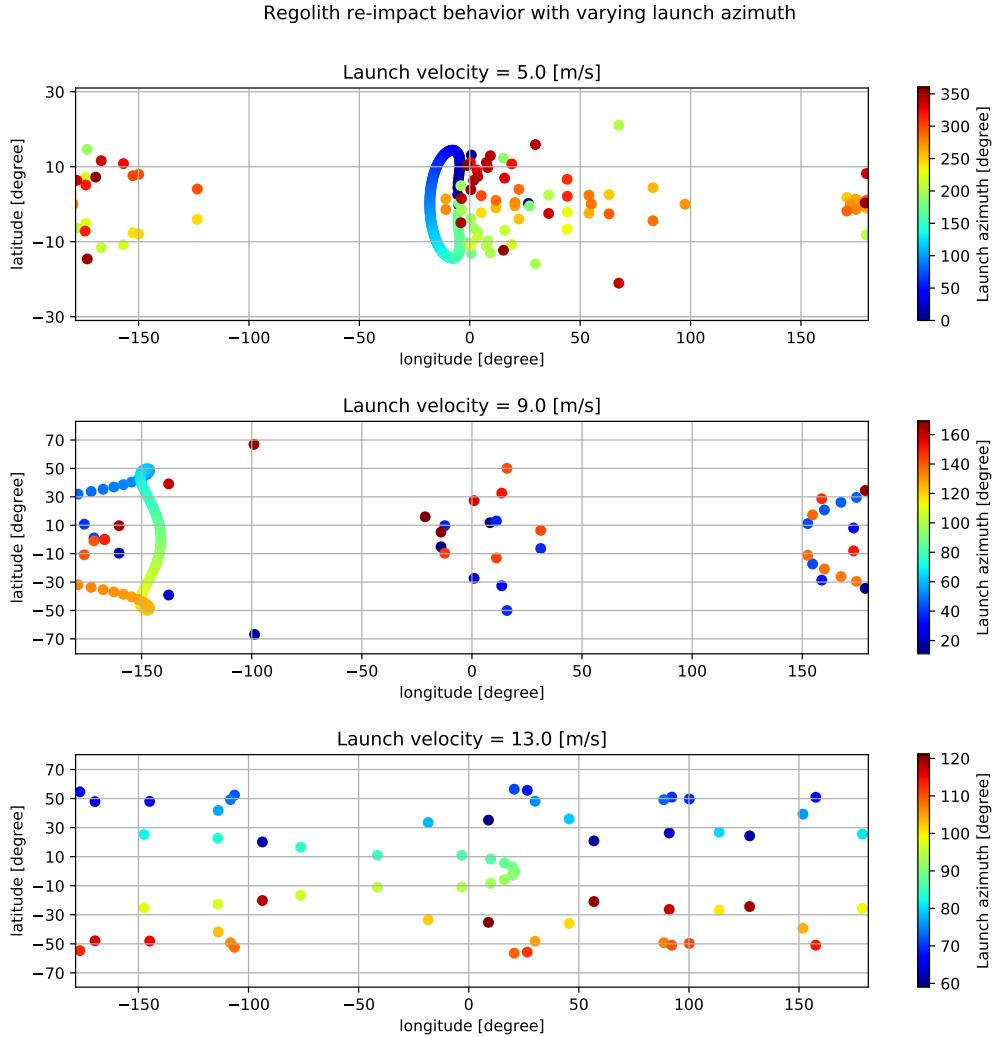


Figure 6.4: Single velocity re-impact location maps for three different velocities. The re-impact locations are color coded based on the launch azimuth.

For the launch velocity of 5 m/s, the re-impact locations corresponding to launch azimuth angles that are against the asteroid's rotational direction (more specifically 7° to 173°)¹, form a distinct curve West of the launch location. These are the particles that soon after launch, re-impact the surface without completing even a single orbit because their energy is reduced significantly enough by the opposing direction of the gravity field. And since the initial launch velocity was inherently small, these particles are not able to reach far away from the launch site either. Launch azimuths 0° and 180° are not directly against or into the asteroid's rotational direction and the corresponding particles complete multiple revolutions before re-impacting the asteroid's surface. Their re-impact location is 0.2° latitude and 26.6° longitude and can be seen in Figure 6.4. The remaining launch

¹Trajectories for launch azimuths 1° to 6° and 174° to 179° get assisted by the fast rotating asteroid which results in them entering escape trajectories

azimuth angles, i.e. the ones that launch the particle in the asteroid's rotational direction², results in a more oddly distributed re-impact locations. The asteroid's rotation assists the launch such that the particles complete one or more orbital revolutions (observed from individual trajectory plots) before re-impacting the surface. This is why for these azimuth angles, the re-impacted particles are not evenly distributed in a smooth curve on the East side of the launch site as the trajectories are not simply ballistic. The re-impact locations in 3D for this launch velocity are shown in Figure 6.5.

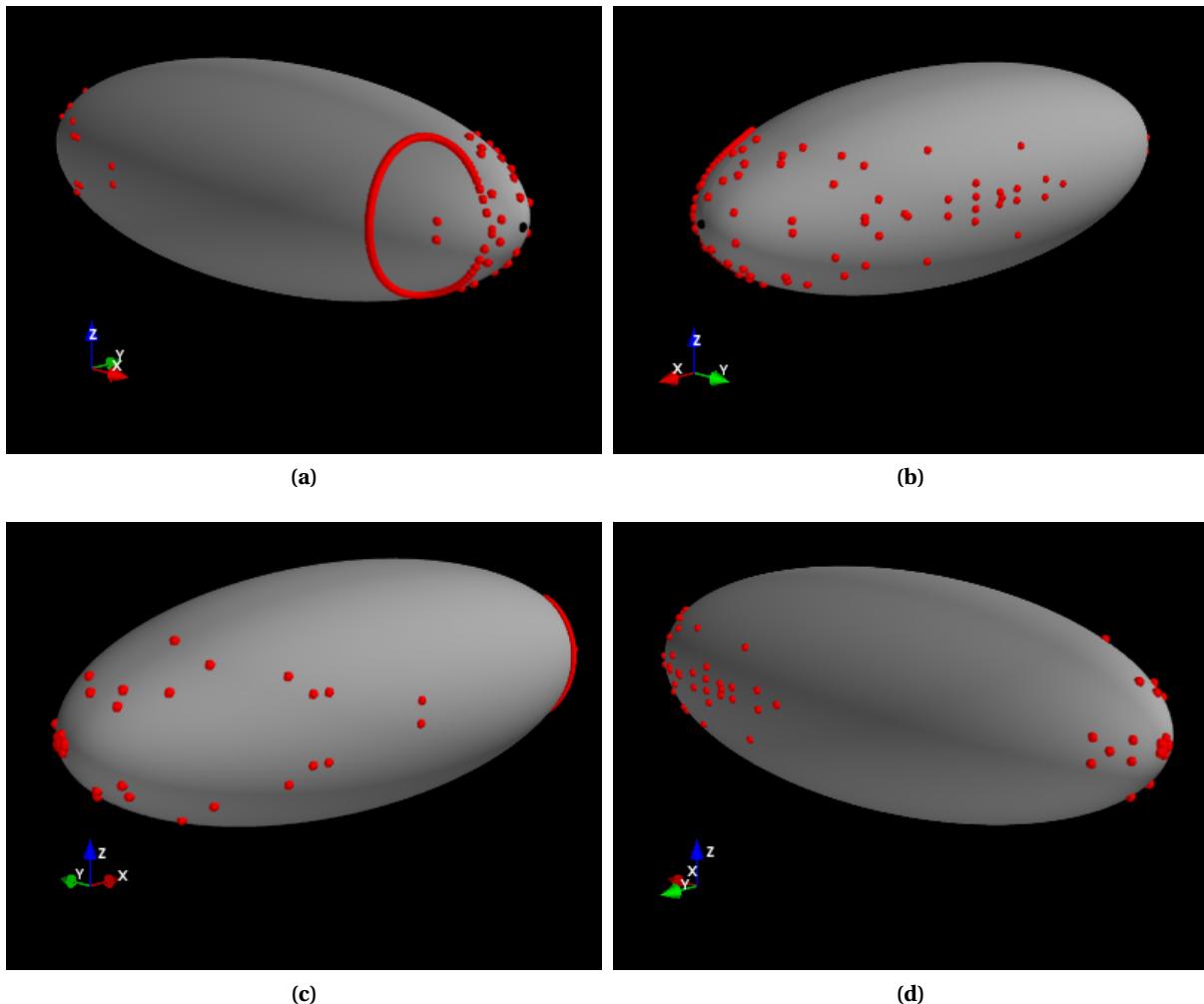


Figure 6.5: 3D view, from four different angles, of regolith re-impact locations for particles launched with a velocity of 5 m/s. The red points represent the re-impact sites and the singular black dot represents the launch site.

Consider the re-impact locations in Figure 6.4 for the launch velocity of 9 m/s. Note that, for this velocity, the launch azimuth angles that result in a re-impact are only the ones which launch the particle in a direction opposite to that of the asteroid's rotation³. Within this range of azimuth angles, the lowermost and the uppermost angles comprise of trajectories that are either ballistic or comprise of one or more orbital revolutions. In case of the ballistic trajectories, we do not see the behavior as we saw for the launch velocity case of 5 m/s where the particles neatly lined up on the West side of the launch site. For the current case, the launch velocity magnitude is inherently

²Not all of these result in re-impact, a few of these also result in an escape scenario but that is irrelevant for the discussion at hand

³Not all the angles within this range correspond to a re-impact situation and some result in an escape as well but the latter is not relevant to the discussion at hand

high. We observed a particle which took the least amount of time to re-impact and also had a ballistic trajectory, and found that the asteroid had rotated a little more than 315° when the particle re-impacted the surface but on the far East end of the asteroid. Since the launch velocity is inherently high, the particle was able to achieve a high enough altitude during which the asteroid had achieved a significant amount of rotation on its own axis, which led to the aforementioned situation.

There is a very distinct collection of re-impact points (the blue-green-orange curve) on the West side of the launch point, in Figure 6.4 for the velocity of 9 m/s and belongs to the azimuth angles from 46° to 136° . The corresponding trajectories are all ballistic in nature and impact the surface relatively sooner after launch. Since the launch velocity is inherently high, unlike in the case of 5 m/s, these particles are able to re-impact a bit further away from the launch site. This is clearly visible in Figure 6.4. For the case of 9 m/s, the 3D view of all re-impact locations is shown in Figure C.6.

Similarly, for the launch case of 13 m/s in Figure 6.4, we first notice that the window or range of launch azimuth angles has reduced further. All particle trajectories, in this case, are ballistic in nature and we do not witness a continuous line of re-impact locations, as opposed to that in the case of 5 and 9 m/s. When launched at 13 m/s, the ballistic trajectories are not short lived, that re-impact the surface soon after launch⁴, which would have allowed a continuous curve on surface to be formed. Since for the current case the launch velocity has a sufficiently larger magnitude, the particles are lofted into even higher altitude trajectories, taking more time to traverse. This is the fundamental difference with the other two velocity cases, as here the asteroid performs several rotations on its axis before the particles re-impact. Thus the ballistic re-impact trajectories for launch velocity of 13 m/s shows similar behavior as that of the multi-orbital revolution cases of 5 and 9 m/s. The re-impact locations in 3D are shown in Figure C.7.

Thus, for lower launch velocities, a continuous curve of re-impact locations can be linked to the launch direction which in our case is the launch azimuth angle⁵. But the same can not be said for higher launch velocities. Another intriguing observation in Figure 6.4 is made for the launch velocity of 5 m/s. We see that the particle launched with azimuth angle of 90° goes further away than, for example, a particle launched with an azimuth angle of 10° . This happens even though the former launch case will experience more resistance in launch since it is directly opposite to the asteroid's rotational direction. The reason for this is simple and becomes apparent if one looks at the trajectory plots shown in Figure C.1.

6.2.3 ESCAPE BEHAVIOR

We now present a brief discussion on the particles that resulted in an escape situation, and do so for the same launch velocities used for explaining the re-impact scenarios. For these launch velocities, Figure 6.6 shows the Hyperbolic Excess Velocity (HEV) for all escaped regolith. As the velocity increases from 5 to 13 m/s, we see that the distribution of particles, based on their HEV, becomes less and less random and more streamlined. There is a link between how the particles are distributed with HEV and the way they escape. To understand this, we take the example of 9 m/s and look at all escape cases from below 100° in launch azimuth. A zoomed-in version of the corresponding data points from Figure 6.6b is shown in Figure 6.7a. In the latter, we can see that for launch azimuths from 0° to 9° , the HEV uniformly reduces and beyond that all other data points are randomly distributed. The distinction between the two regimes becomes apparent when we look at the oscu-

⁴Soon after launch, here, refers to the fact that the ballistic trajectories re-impacted the surface before the asteroid could complete even a single rotation on its own axis

⁵The launch declination angle for our simulation was kept constant

lating energy and eccentricity for all these data points, as shown in Figure 6.8.

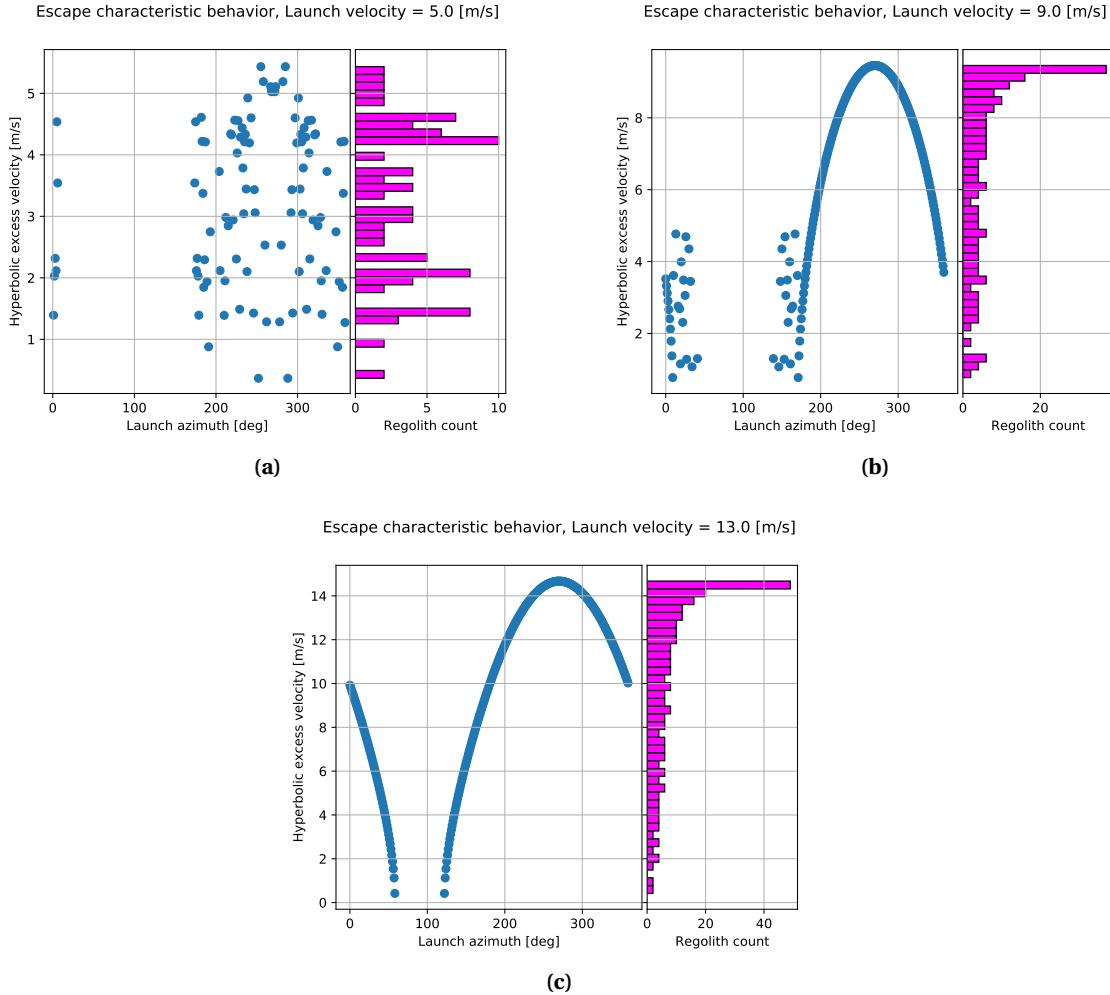


Figure 6.6: Hyperbolic escape velocity versus launch azimuth for three different launch velocity cases. The histogram bars in each plot depict the number of regolith for the various hyperbolic excess velocities on the y-axis.

So for the data points with uniformly reducing HEV values, we observe that the energy never becomes negative and the eccentricity always stays above 1.0, which means that the corresponding regoliths are on an escape trajectory immediately after launch. It doesn't take a gravity assist from the fast rotating asteroid to put these regolith on an escape route. This is an important distinction against the randomly distributed data points in Figure 6.7a. For the latter, from the eccentricity and energy curves, we can witness that the particles briefly enter bounded orbits around the asteroid but eventually escape with a sudden increase in energy (becoming positive) and eccentricity (going above 1.0). This happens because, after launch, the particle trajectories get gravity assist from the asteroid that is sufficient to put the particle on an escape route. What was shown here for an example of 9 m/s was not an isolated case and was found to be true for other launch velocities as well, but their results are not shown here for brevity. Also, this behavior stands true for data points with uniformly increasing HEV values such as in the case of 9 and 13 m/s. Thus, just by looking at the HEV plot, we can now determine which launch azimuths led to an immediate escape and which ones required gravity assist from the asteroid to escape. The trajectory plots for a few launch azimuths that highlight this distinction is shown in Figure 6.7b.

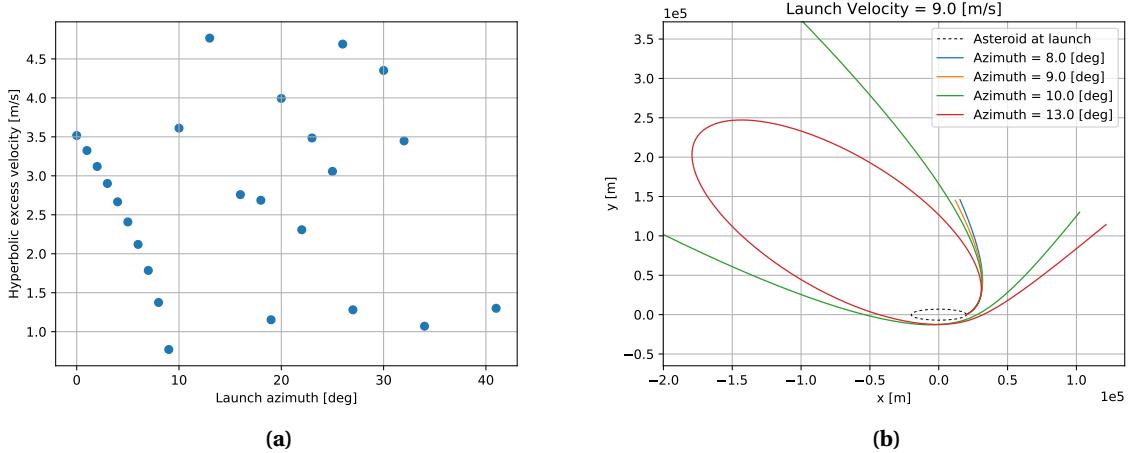


Figure 6.7: (a) Zoomed-in data points for azimuth angles below 100° from Figure 6.6b. (b) Example escape case trajectory plots for select few launch azimuths and a launch velocity of 9 m/s.

6.2.4 SENSITIVITY TO LAUNCH CONDITIONS

In this section, we will briefly highlight how regolith motion is sensitive in terms of their final fate, i.e., escape or re-impact. Recalling from Section 6.1, the launch site and launch declination angle remain constant in this simulation exercise. We vary only the launch azimuth angle and the launch velocity of the regolith while lofting them. Hence, we judge the sensitivity in terms of these two initial elements.

Figure 6.9 highlights the difference between two trajectories that were launched with the same azimuth angle but differently launch velocities. An escape situation is encountered when the particle is launched with a velocity of 4 m/s while the same particle re-impacts when launched with a velocity of 5 m/s. With the resolution of launch conditions established in general for our simulation, and for every other launch condition being the same, the particle is sensitive to either of the final fates with 1 m/s of a difference in the launch velocity.

Figure 6.10 highlights the difference between two trajectories when the launch velocity is the same, i.e. 10 m/s, but the launch azimuths are different. So at azimuth 41° , the particle escapes while at 42° the particle re-impacts the asteroid. Thus, for all other launch conditions being the same, the particle is sensitive to either of the final fates to a 1° launch azimuth resolution.

The current discussion, albeit short, was to see if the sensitivity (in terms of final fate acquired by regolith) had a magnitude equal to the resolution of varying the launch conditions in the current simulation scenario. This was found to be true, but, from all the results that we have seen so far and looking at how blurry the line in distinguishing the two fates, a similar sensitivity behavior can be expected if the resolution of launch conditions is lowered.

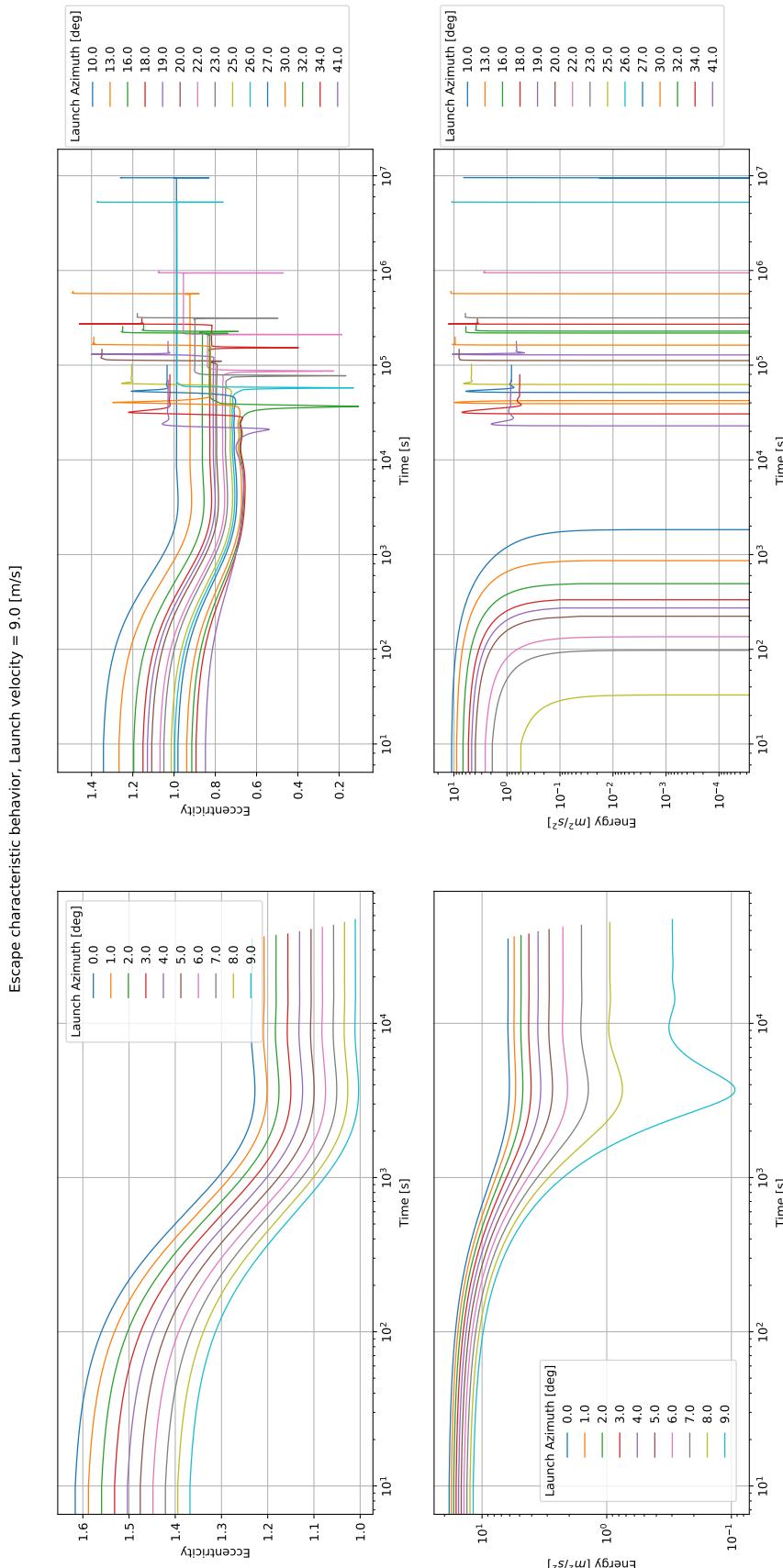


Figure 6.8: Osculating energy and eccentricity for all escape cases as shown in Figure 6.7a. The energy plot is shown in an unorthodox way by using a logarithmic scale to easily distinguish energies that stay positive for the entire duration until escape. If the energy becomes negative, then on this scale, the corresponding curve appears to be going straight downwards in the plot (since a negative value is not defined on the log scale). Similarly, if the energy becomes positive from a negative value earlier, then the curve appears to be coming upwards straight (again from some hypothetical negative value at the bottom of the log scale).

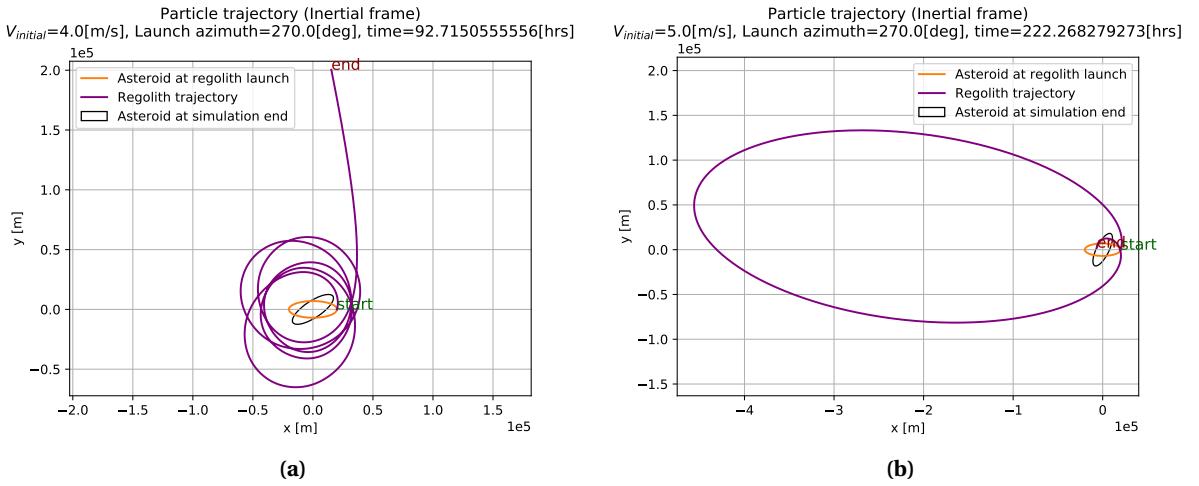


Figure 6.9: Regolith trajectory, expressed in the AIF, for particles launched with azimuth 270° and velocity of (a) 4 m/s and (b) 5 m/s.

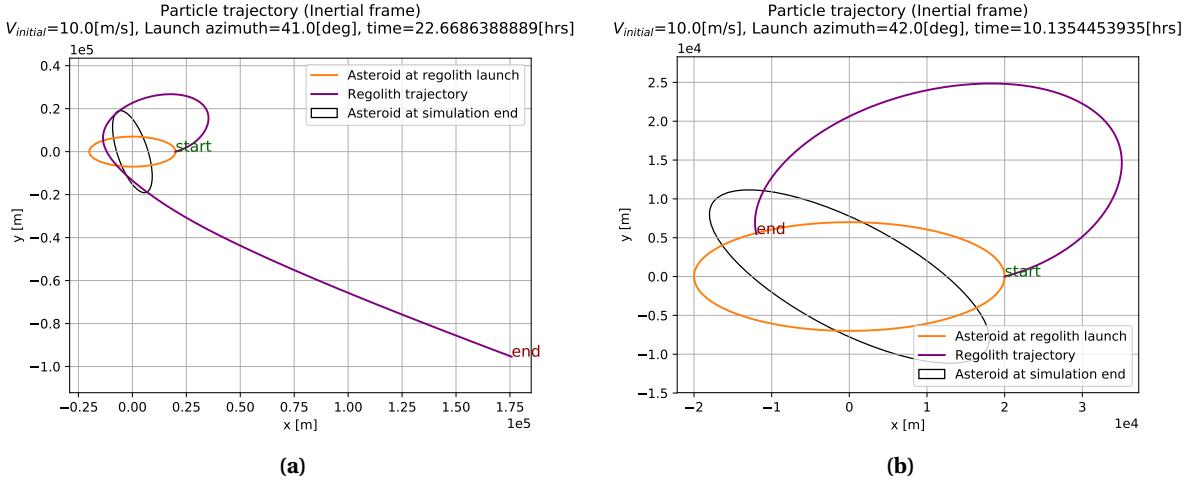


Figure 6.10: Regolith trajectory, expressed in the AIF, for particles launched with velocity 10 m/s and azimuth of (a) 41° and (b) 42°.

7

DYNAMICS WITH SOLAR PERTURBATIONS

The discussion on results so far have not accounted for any Solar perturbations in the simulations. In this chapter, however, we will see how adding STBE and SRP perturbing accelerations, in addition to the non-uniform gravity field of the asteroid, affects the motion of regolith and if any capture orbits are obtained due to them.

7.1 REGOLITH SIZE AND DENSITY

Before we can present the results of any simulation, we need to justify the size and density of different regolith types used. We used two different density values and two different grain radii and thereby used 4 different area-to-mass ratios. The selection of regolith types was made by using asteroid Eros as an example. The different area-to-mass ratios are mentioned in Table 7.1.

7.1.1 DENSITY SELECTION

The material with a density of 3.2 g/cm^3 is low-density Olivine (Magnesium Iron Silicates) and the one with 7.5 g/cm^3 is Iron-Nickel alloy (Garcia-Yarnoz et al. 2014). We have chosen these two types of materials based on the surface composition analysis of asteroid Eros, an S-Type asteroid, from the NEAR-Shoemaker data. S-Type asteroids, from reflectance spectral analysis, are commonly known to have minerals like Olivine, Pyroxene, and Fe-Ni (Iron-Nickel) metal (Nittler et al. 2001). Thermo-spectral analysis of regolith on Eros reveals that it is rich in Olivine and is found to be more abundant than Pyroxene (McCoy et al. 2001). The mineral Olivine has also been discovered on Itokawa, another S-Type asteroid, through transmission electron microscope analysis of samples returned by the Hayabusa spacecraft (Keller et al. 2014).

Eros also contains Fe-Ni but it is significantly separated from the Silicates (Olivine and Pyroxene) within the regolith (Nittler et al. 2001). Evans et al. 2001 analyzed elemental composition of NEAR-Shoemaker's landing site on Eros, based on which, it presents several arguments for relatively lower abundance of Fe (Iron) on the surface of Eros. One of the arguments hypothesizes that different grain sizes and density of Fe-Ni from Olivine could have resulted in the metal to get separated from the Silicates, either spatially or for it to sink down in the lower depths of the regolith. In light of this, we are considering regolith comprising of only Olivine and Fe-Ni, to distinguish between their orbital behavior and final fate upon being lofted from the surface of an asteroid such as Eros.

7.1.2 SIZE SELECTION

Veverka et al. 2001a analyzed high resolution surface images of Eros captured by NEAR-Shoemaker on a low-altitude flyover. It argued the build-up of a heterogeneous and complex regolith that com-

prised of material ranging from fine particles all the way up to metre-sized ejecta blocks. Veverka et al. 2001a argues that while there is an abundance of large ejecta blocks across the surface, the much finer regolith occupies mostly the low-lying topographies, i.e., inside large craters on the surface of Eros. The latter was termed as ponded deposits. Robinson et al. 2001 argues, from high resolution images (1.2 cm per pixel) of ponded deposits at Eros, that the grain size of regolith would be around 1.0 cm or below.

Thus based on this extreme spectra of regolith composition at Eros, we shall also consider regoliths with varying densities and grain radii (each grain is assumed to be spherical). These are listed in Table 7.1. The particles are listed in decreasing order of area-to-mass ratio.

Table 7.1: Particle Area-to-Mass ratios

Code	Particle radius [cm]	Density [g/cm ³]	Area-to-Mass ratio [m ² /kg]
LoGSP-1	1.0	3.2	0.0234
LoGSP-2	1.0	7.5	0.01
LoGSP-3	5.0	3.2	0.0047
LoGSP-4	5.0	7.5	0.002

7.2 SIMULATION SETUP

The initial conditions for lofting each type of regolith are varied in the same manner and are mentioned as follows. The asteroid revolves around the Sun in an equatorial circular orbit at a distance of 1.0 Astronomical Unit (AU). Four different initial Solar phase angles were considered for the simulation – 45.0°, 135.0°, 225.0° 315.0°, to account for the four different quadrants where the Sun could be with respect to the asteroid.

For each case in Table 7.1, a total of 72 particles¹ were launched from the surface of the asteroid, each in a different direction. The launch declination angle was kept constant at 45.0° for all the particles. The launch azimuth was varied at a resolution of 5.0° starting from 0.0° all the way up to 355.0°. Each particle was launched, in their specified direction, with different velocities ranging from 1.0 m/s to 16.0 m/s (measured with respect to the ARF) at a resolution of 1.0 m/s. So basically, every combination of an initial Solar phase angle, initial launch azimuth, and initial launch velocity corresponds to a unique trajectory for a single particle of a given Area-to-Mass ratio; Thus amounting to a total of 4608 unique trajectories for each regolith type.

We choose three different launch sites this time; the first being the longest edge of the CDE at latitude and longitude both 0°; the second being the leading edge at latitude and longitude both 45°; and finally the third being the trailing edge of the asteroid at latitude 45° and longitude –45°. However, for a detailed analysis on the effects of Solar perturbations on regolith motion (discussed in Section 7.3), we will consider only the longest edge launch site. The remaining launch sites will be used later on to present more general and statistical results (Section 7.4) that characterize the final fate and orbital motion of regolith based simply on different grain size and density, for the respective launch sites.

¹Recall that in the previous chapter, the number of particles were launched at an azimuth resolution of 1° which resulted in 360 particles being launched

7.3 SOLAR PERTURBATION ANALYSIS

We now present the orbital motion analysis for one of the regolith types, particle LoGSP-1, when lofted from the longest edge of the asteroid. We choose this particular regolith type since it offers the maximum area-to-mass ratio. A larger value for area-to-mass ratio means a relatively larger effect of SRP on the regolith which makes it more interesting since for a detailed analysis we want to see how SRP (as well as STBE in general) affects the orbital motion of regolith. Presenting a similar detailed analysis for all particle types will not be practical for the purposes of this report since it will become repetitive.

7.3.1 COMPARISON WITH UNPERTURBED SCENARIO

We'll begin with a very brief comparison of the final fate characteristics of regolith from when the Solar perturbations were included in our simulation, with the case when they were excluded. Note that, when we say unperturbed, it simply means the exclusion of the SRP and STBE and not the non-uniform gravity field. Since the only two outcomes in the unperturbed scenario were escape and re-impact, we compare them with the results from the simulation with perturbations.

We begin by first comparing the re-impact behavior. Figure 7.1a shows the re-impact locations for regolith launch in an environment with no Solar perturbations, with three different launch velocities as shown in the plot. The plot is the same as Figure 6.4 except that the launch azimuth resolution is 5° instead of 1° . Similarly, Figure 7.1b shows the re-impact locations for regolith simulated in Solar perturbations. The two cases are very similar to each other. The location and shape of the re-impact points, forming an arc on the West of the launch site for launch velocities of 5 m/s and 9 m/s, remains the same for both the perturbed and unperturbed scenario. In those cases, the trajectories are short lived and ballistic and the perturbations don't get enough time to accumulate and change the trajectory. For all other re-impact points, the numbers are smaller for the perturbed case and the corresponding re-impact locations are different from the unperturbed counterpart. In general, the re-impact map for the perturbed scenario shares a good resemblance with the unperturbed one. This aforementioned features of the re-impact locations was found to be similar for results from all four initial Solar phase angles considered in the simulations, but the plots for each is not shown in this report for brevity.

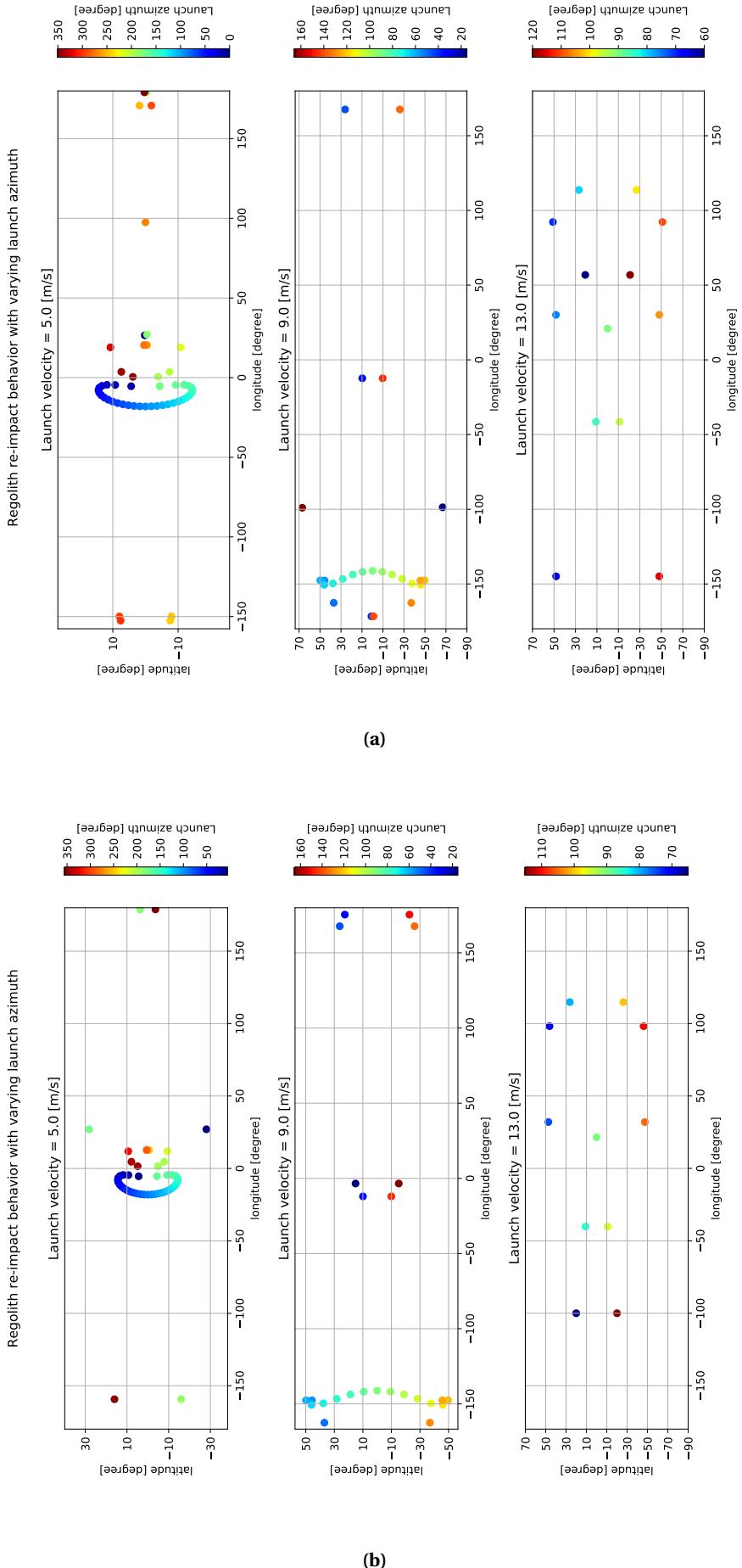


Figure 7.1: Re-impact locations for three specific velocities, where for each velocity, the launch azimuth varies with a resolution of 5° . Out of the two figures, (a) shows the results for the unperturbed scenario and (b) shows the results for the case with the perturbations. For the latter, the initial Solar phase angle is 45° and particle code is LoGSP-1.

Now we will look at the escape behavior. Recall how in Section 6.2.3 we discussed the link between the escape trajectories and distribution of data points in the HEV versus launch azimuth plot. It was found that a continuous and streamlined distribution of data points implied that the particles had escaped immediately without using any gravity assists while this wasn't true for the remaining data points that were distributed chaotically. This correlation was found to be true even in the case of the perturbed simulation. We show an example case for launch velocity of 9 m/s and for launch azimuths from 0° to 20° in steps of 1° , in Figure 7.2. What we see in the figure is that for all launch azimuths below 10° , the particles are on an escape trajectory immediately after launch. For escaping particles with launch azimuth above 10° , the particles required gravity assists from the asteroid and involved taking one or more revolutions around it before finally escaping.

So far the behavior and the explanation for it is the same as that in Section 6.2.3, however, there is a critical case witnessed here at launch azimuth 10° . From the discussion earlier and from Figure 7.2a, one would say that this particle should already be on an escape trajectory as it is aligned with the smoothly reducing HEV data points of lower azimuths. But that is not the case. Also note that the HEV for it is 0 m/s and from Figure 7.2b we see that at the time of escape the trajectory has an eccentricity of 1.0. Both these facts confirm that the osculating trajectory finally becomes parabolic at the instance of escape. This is perhaps why this particular regolith doesn't share the same traits with any other launch azimuth, because in every other case the escape trajectory is hyperbolic.

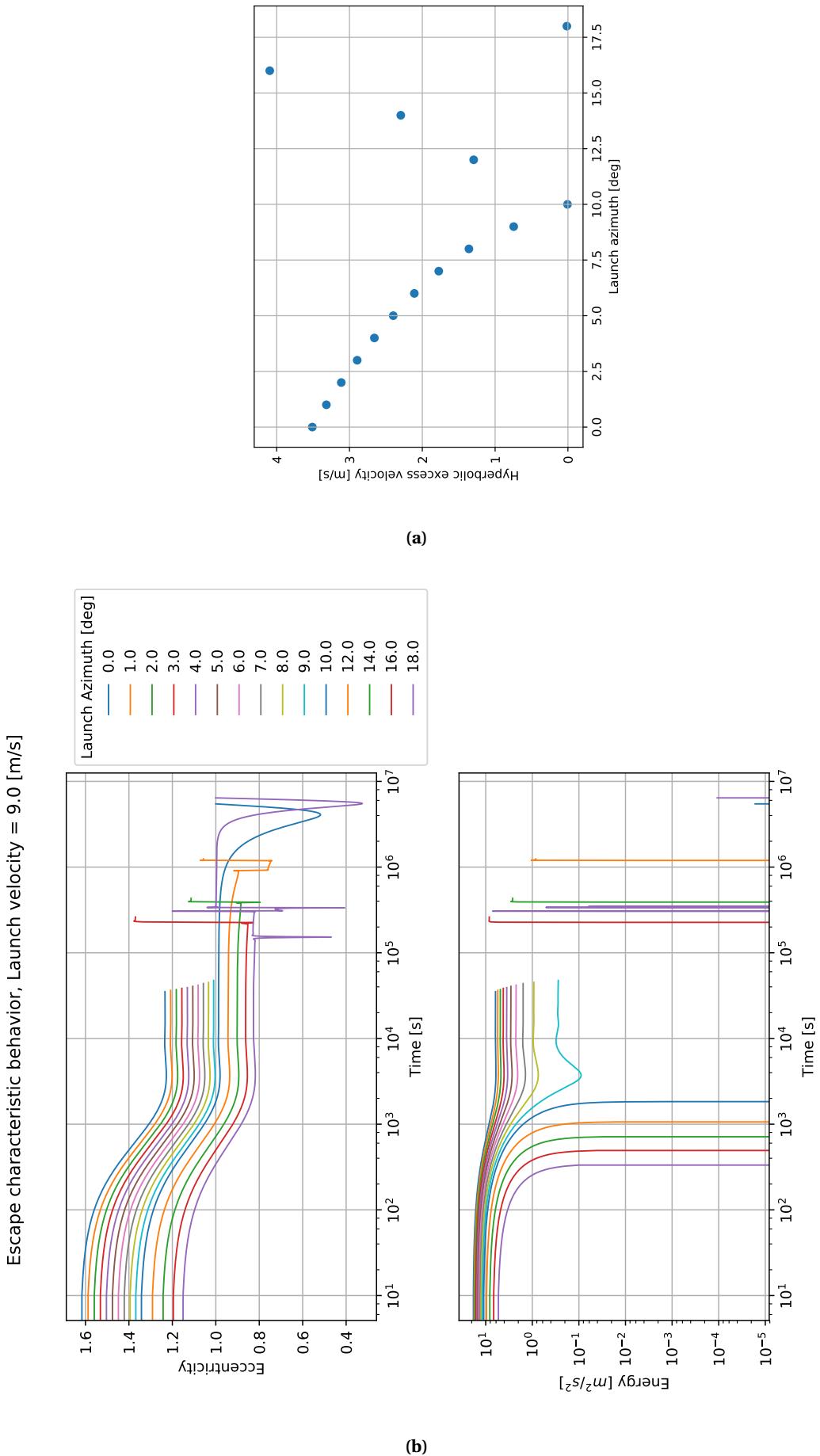


Figure 7.2: Escape behavior for regolith launched with a velocity of 9 m/s; launch azimuths shown here range from 0° to 20° in steps of 1°, which is sufficient to discuss the topic at hand. Out of the two figures, (a) shows the HEV data points and (b) shows the energy and eccentricity curves for the corresponding cases. For the latter, the initial Solar phase angle is 45° and particle code is LoGSP-1.

7.3.2 FINAL FATE GENERAL CHARACTERISTICS

Figure 7.3 gives a distribution of particles for each of the three different final fates for the regolith i.e. capture, re-impact, and escape, for different initial launch velocities and initial Solar phase angles. Irrespective of the initial Solar phase, initial launch velocities from 1.0 m/s to 3.0 m/s results in particles launched in all directions to eventually re-impact the asteroid's surface. Similarly, for initial launch velocities ranging from 14.0 m/s to 16.0 m/s, we see that the particles always manage to escape the gravitational attraction of the asteroid². Launch velocities from 4.0 m/s to 13.0 m/s show a mixed behavior and the final fate distribution trend does not vary drastically for different initial Solar phase angles. Looking at absolute numbers, for all Solar phase angles combined, about 54.9% cases result in an escape situation, 44.8% cases result in re-impact and a mere 0.2% cases result in a capture scenario. Thus, the number of capture cases is extremely small relative to the other two final fates.

For initial Solar phase of 225.0 [deg], there are no cases of regolith being captured in orbit around the asteroid. All capture cases, arranged in order of increasing launch azimuth angle, are listed in Table 7.2. It is interesting to note that all capture cases result from when the particle is launched in a direction which is against the direction of rotation of the asteroid.

Table 7.2: Initial conditions that resulted in temporary orbital capture of regolith around the asteroid. Particle code LoGSP-1.

Index	Launch azimuth [deg]	Launch velocity [m/s]	Initial Solar phase angle [deg]
1	5.0	5.0	315.0
2	10.0	9.0	135.0
3	15.0	8.0	45.0
4	45.0	12.0	45.0
5	45.0	10.0	315.0
6	135.0	12.0	45.0
7	135.0	10.0	315.0
8	165.0	8.0	45.0
9	170.0	9.0	135.0
10	175.0	5.0	315.0

The capture cases which represent symmetry in terms of the launch azimuth angle are highlighted with the same color in Table 7.2. This symmetric behavior results from the combination of two factors. First, the Sun's motion relative to the asteroid is not in an inclined plane, and secondly, the particles are launched from the equatorial tip of the ellipsoid shaped asteroid, which is a point of symmetry on the ellipsoid. The capture cases are discussed in detail, later in the report.

²There is one exception though, at 14.0 m/s and for launch azimuth of 90° and Solar phase angle of 315°, the particle re-impacts

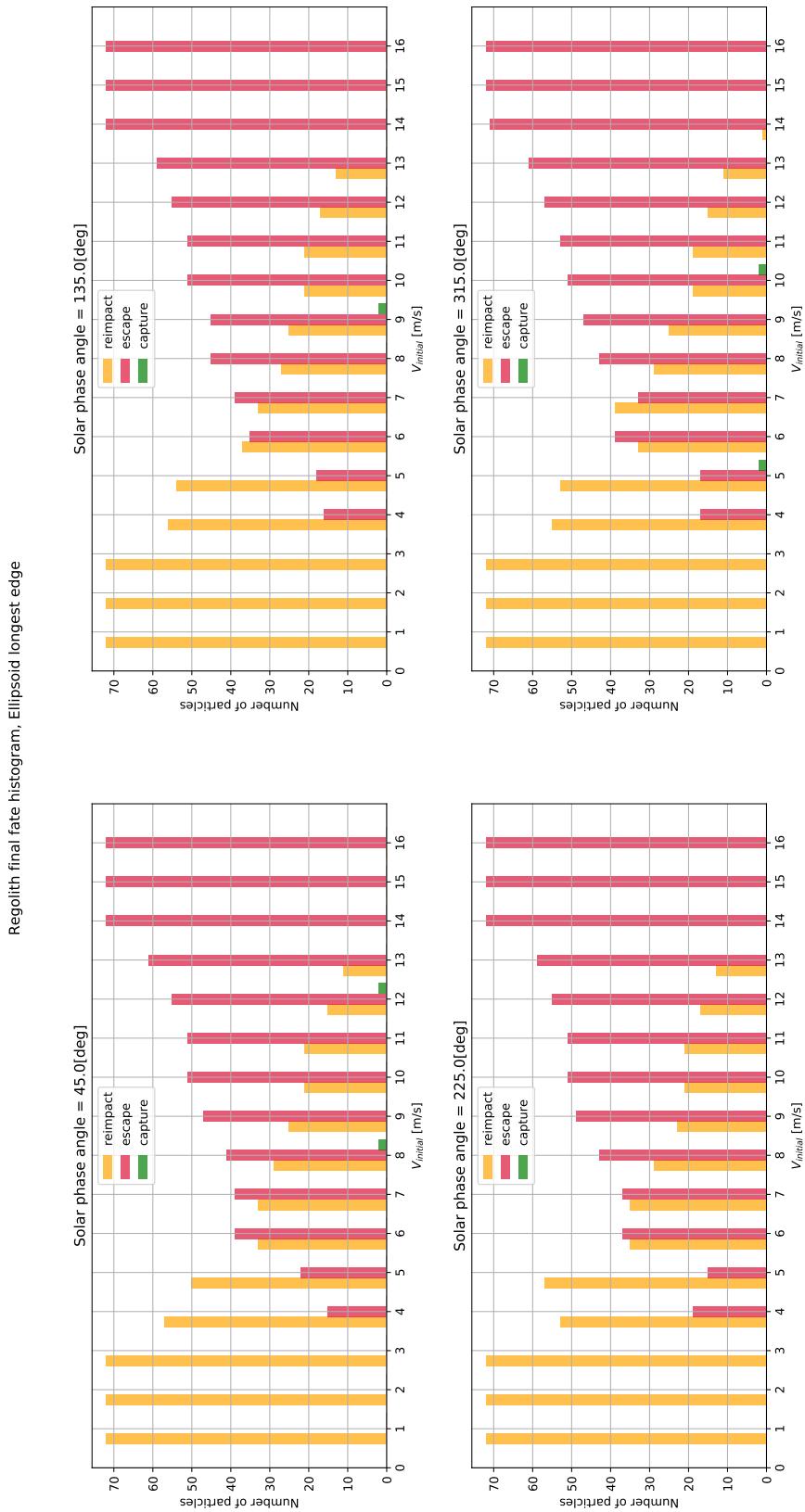


Figure 7.3: Histogram showing the number of particles that re-impact, escape, or get captured around the asteroid, for different initial launch velocities. Particle code LoGSP-1.

Figure 7.4 depicts the surface distribution of regolith that re-impacts the surface when launched from the same location with different velocities and different initial Solar phase angles. The launch location is in the centre of the map, Latitude and Longitude 0.0° . The re-impact locations remains unchanged, irrespective of the initial Solar phase angles, for all particles launched with the relatively lower velocities (1 m/s to 4 m/s). Again for the mid-range velocities, i.e. 5 m/s to 9 m/s, the re-impact locations for the particles that are launched in the direction opposite to the asteroid's rotation remains unchanged for different initial Solar phase angles. This can be observed in the arc shaped re-impact location patterns West of the launch location for those velocities. The distribution pattern, for all launch velocities and initial Solar phases, is also symmetric about the equator. Again, the reason for this is the same as mentioned earlier for the symmetry in capture cases in Table 7.2. In general, keeping the launch direction and velocity constant, we see that the distribution of re-impact locations does not change drastically with varying initial Solar phase angles. This is much easily observed in a plot of Range from the launch direction to the re-impact point versus launch azimuth for different velocities as shown in Figure 7.5.

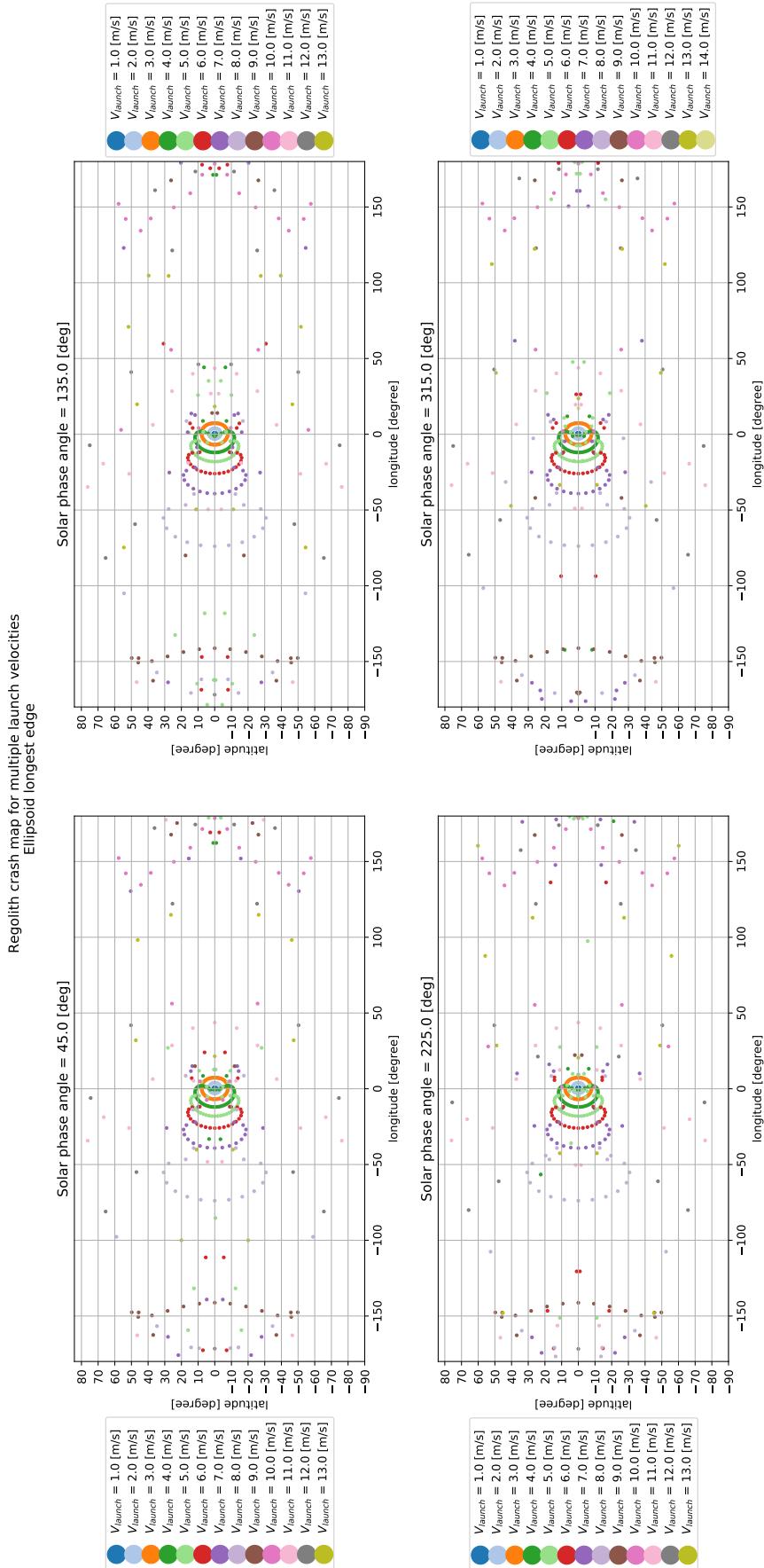


Figure 7.4: Surface distribution of re-impacted regolith for different launch velocities. The launch location is latitude: 0°, longitude: 0.0°. Particle code LoGSP-1.

We haven't shown the range to re-impact point plots in Figure 7.5 for all launch velocities because the intention here is to show the qualitative behavior, which can be achieved by considering only a subset of the launch velocities that result in a re-impact scenario. The very first thing we observe is that as the launch velocity increases, the range of launch azimuth over which the regolith re-impacts the surface reduces because a higher velocity allows the regolith to enter a higher orbit (as it attains a relatively higher energy) and reduces the probability of a re-impact. Even as the velocity increases, we see that the azimuths that result in a re-impact are the ones in which the regolith is launched in a direction that is opposite to the asteroid's rotation direction. This makes sense since the regolith's energy would be reduced the most in this scenario compared to all other launch directions, thereby increasing the chances of a re-impact.

Now the primary purpose of the plots in Figure 7.5 (combined with Figure 7.4) is to depict the qualitative effect of Solar perturbations, for varying initial Solar phase angles, on the re-impact behavior of regolith compared to the case when no Solar perturbations are considered. For launch velocities of 4.0, 7.0 and 10.0 m/s, we see that the Solar perturbations do not affect the re-impact location for cases when the particle is launched in directions opposite to that of the asteroid's rotation. However, we do see few exceptions to the former statement, most noticeably in the case of 7.0 m/s. But for the majority of cases where the re-impact location remains unchanged, we see from Figure 7.6, that these particles spend less than 3.0 hours in orbit which is not enough time for the Solar perturbations to act and have any significant impact on the dynamics of the particles. So in essence this is what's happening here - Particles when launched in a direction that is opposite to that of the asteroid's rotation, even at relatively high velocities such as 10.0 m/s, loose enough energy to stay in a relatively lower orbit (see Figure C.30) where the gravitational force of the asteroid is significantly stronger than any of the Solar perturbations and as the particle spends a very short time in orbit before re-impact, the Solar perturbations do not get enough time to affect the particle's orbit and hence the particle re-impacts the same location as it would have when no Solar perturbations were considered in the simulation. For the lower launch velocities of 4.0 and 7.0 m/s, the differences in re-impact locations are more pronounced when the regolith is launched in the same direction as that of the asteroid's rotation. Particles gain relatively higher energy in this case, enter a higher orbit and spend enough time in there for the Solar perturbations to affect its motion. For the case of the launch velocity of 13.0 m/s in Figure 7.5, the velocity is high enough such that the particle does not loose enough energy when launched opposite to the asteroid's rotational direction and is able to enter a relatively higher orbit (see Figure C.30) and stay there for a relatively longer time, as seen in Figure 7.6, which results in the Solar perturbations affecting the orbital motion and eventually the re-impact location of the regolith.

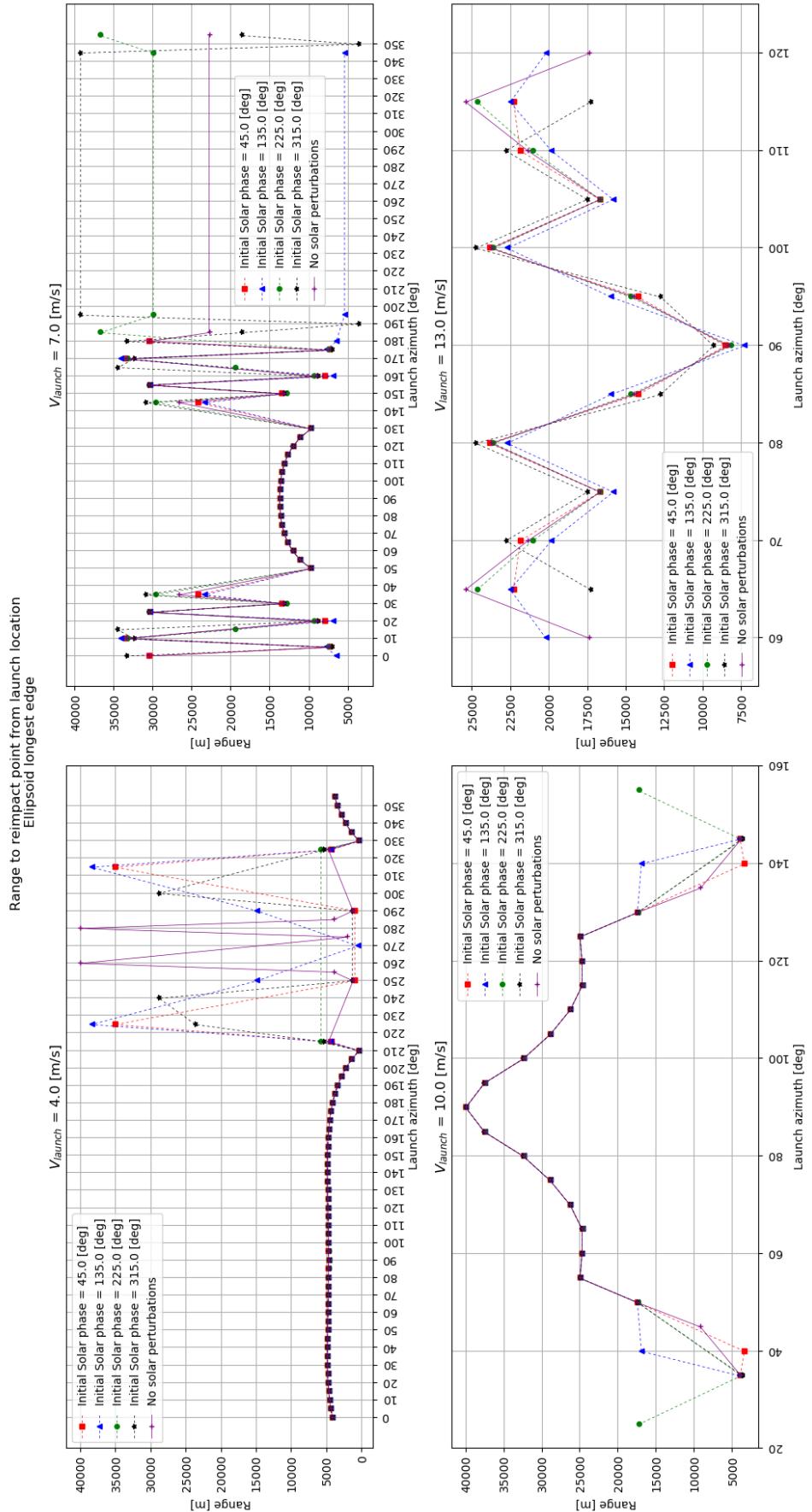


Figure 7.5: Range to re-impact location from the launch point for different velocities. Particle code LoGSP-1.

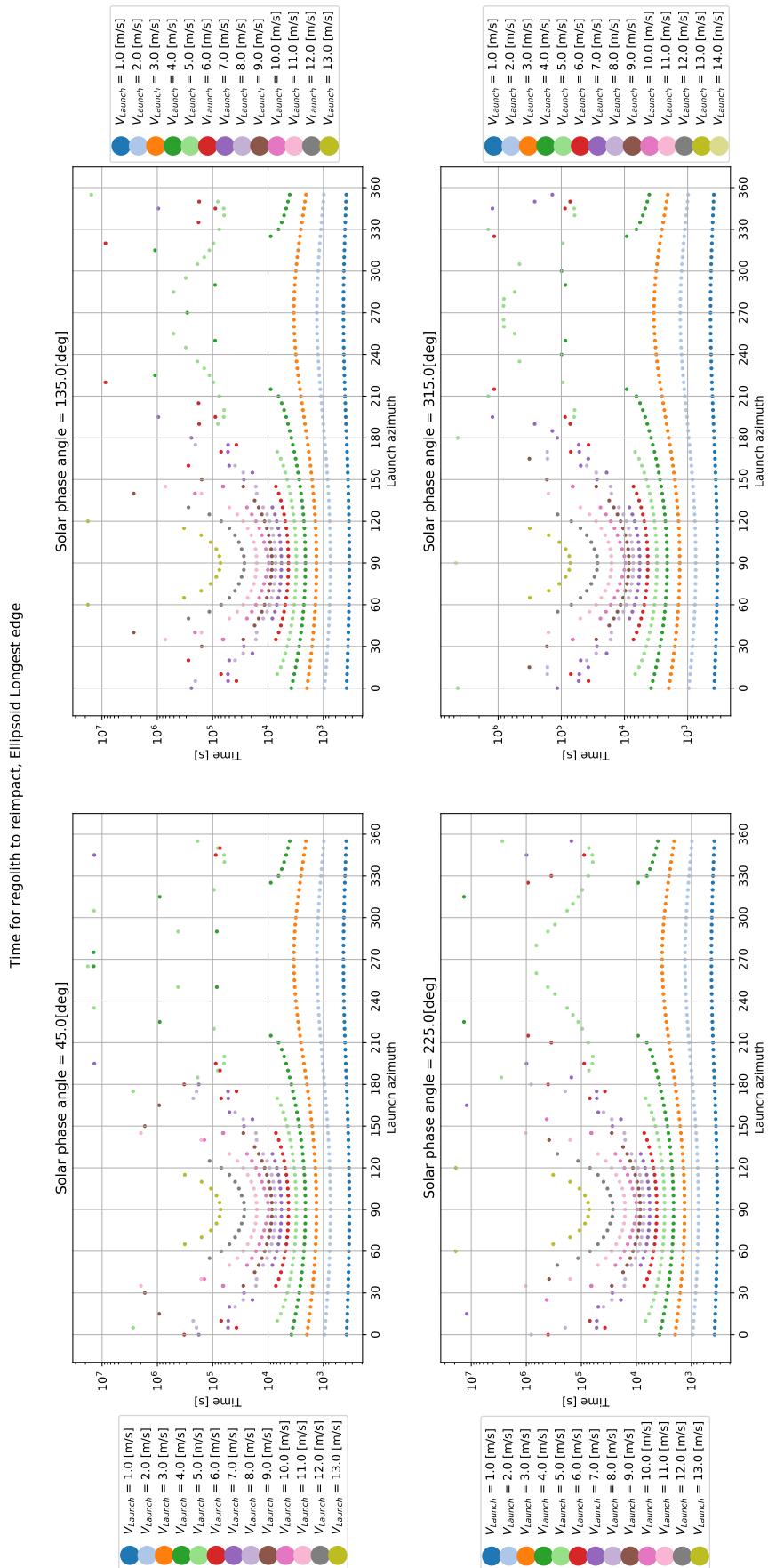


Figure 7.6: Time taken by regolith at different velocities and launch directions to re-impact with the surface of the asteroid. Particle code LoGSP-1.

7.3.3 CAPTURE ORBIT ANALYSIS

We shall now look at the cases where the lofted regolith gets (temporarily) captured in orbit by the asteroid. The initial conditions for all capture cases, for the current particle size and density, were mentioned earlier in Table 7.2. Figure 7.8 depicts the progression in orbital range of the temporarily captured regolith. The straight lines in the plot are used to mark the different altitude regimes. These are the Low Altitude Orbit (LAO), Medium Altitude Orbit (MAO), High Altitude Orbit (HAO), Ultra-High Altitude Orbit (UHAO), and Extremely-High Altitude Orbit (EHAO). These altitude regime definitions are not from well defined standards, but instead were arbitrarily chosen as integer multiples of the longest semi-major axis, α , of the tri-axial ellipsoid shaped asteroid. The definition for these altitude regimes is given in Table 7.3.

Table 7.3: Altitude regimes and their definitions

Altitude regime	Definition
LAO	Asteroid surface to $2 \times \alpha$
MAO	$2 \times \alpha$ to $3 \times \alpha$
HAO	$3 \times \alpha$ to $5 \times \alpha$
UHAO	$5 \times \alpha$ to $7 \times \alpha$
EHAO	Above $7 \times \alpha$

The purpose of plotting data as shown in Figure 7.8 was to look for any patterns or periodicity, if they existed, and to see if particles in temporary capture scenario remain closer to the asteroid or further away from it. The symmetry as explained for initial conditions mentioned in Table 7.2 can also be seen in Figure 7.8, for example, regolith launched with velocity of 8.0 [m/s] and launch azimuth of 15.0 [deg] (shown by the purple curve in the top plot in Figure 7.8) shows the same behavior as that of regolith launched with the same velocity and 165.0 [deg] launch azimuth (shown by the green curve in the bottom plot in Figure 7.8). Another thing we see from the plot is that the captured regolith stay in the higher altitude regions for most part and only briefly do they fall within the MAO and LAO region. We shall now take a look at a few cases from Figure 7.8 in a bit more detail to understand the effect of Solar perturbations by comparing these cases with their unperturbed counterparts.

Of all the cases shown in Figure 7.8 or Table 7.2, the one with a launch velocity of 10.0 [m/s] and launch azimuth of 45.0 [deg] results in a re-impact scenario when Solar perturbations are omitted but the same initial conditions lead to a temporary capture orbit when perturbations were added for an initial Solar phase angle of 315.0 [deg]. Every other initial condition for the capture cases had otherwise resulted in an escape situation when simulations were conducted without the Solar perturbations. The 3D trajectory plot in two different views for the former case are shown in Figure 7.9 (see Figure C.31 also for the 3D trajectory representation in body fixed frame). The 2D trajectory for the same is shown in Figure 7.10 in inertial frame and in Figure C.32 in the asteroid centric rotating frame or the body frame. The web-link or URL for the trajectory animation of the particle (in inertial frame and in XY plane only) can be found in Figure 7.7.



Figure 7.7: 2D trajectory animation (XY Plane) of capture regolith for case number 5 in Table 7.2. Particle code LoGSP-1. Scan the QR code to view the animation or use the following web-link: <https://youtu.be/oZDhDo5CIsk>

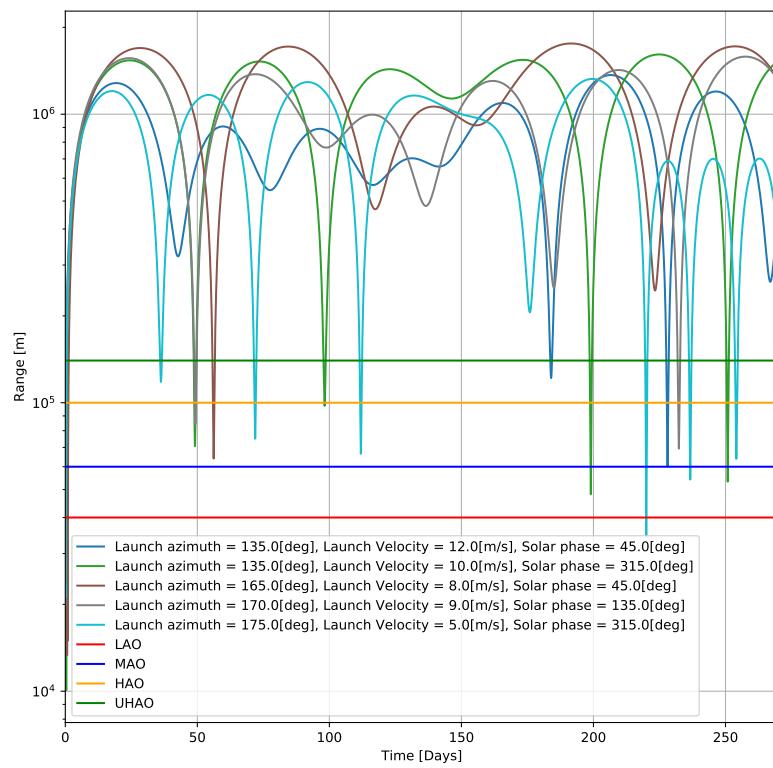
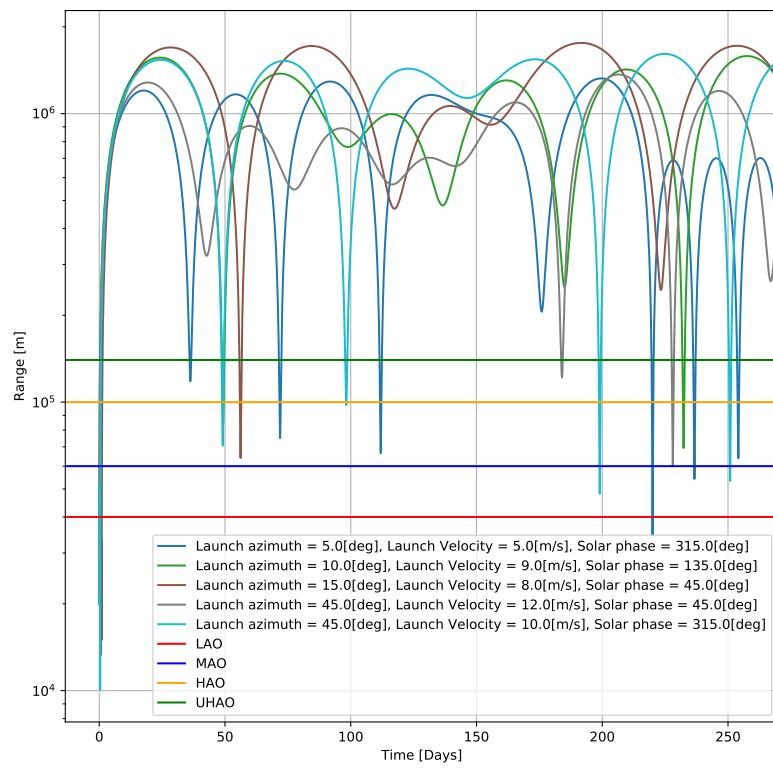
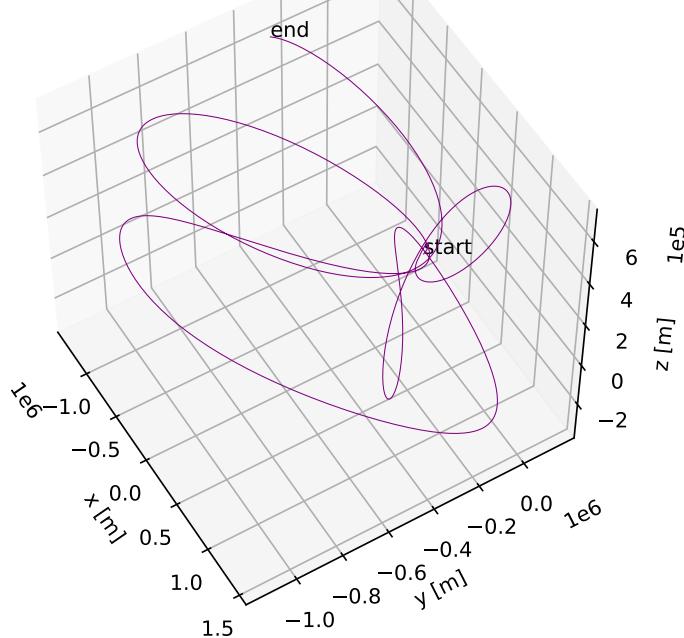


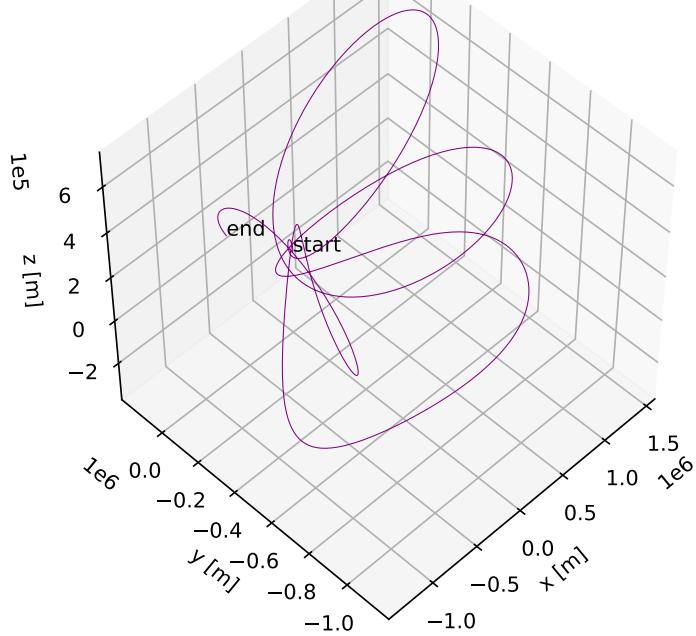
Figure 7.8: Orbital range progression with time for temporary capture scenarios. Particle code LoGSP-1.

Ellipsoid longest edge, Inertial frame trajectory
 $V_{\text{launch}} = 10.0 \text{ [m/s]}$, Launch azimuth = 45.0 [deg], Solar phase = 315.0 [deg]
Time = 0.0 to 270.0 [days]



(a)

Ellipsoid longest edge, Inertial frame trajectory
 $V_{\text{launch}} = 10.0 \text{ [m/s]}$, Launch azimuth = 45.0 [deg], Solar phase = 315.0 [deg]
Time = 0.0 to 270.0 [days]



(b)

Figure 7.9: 3D inertial frame trajectory of capture regolith for case number 5 in Table 7.2 in two different viewing angles.
Particle code LoGSP-1.

Note that in the trajectory animation in Figure 7.7 (and any other animation included henceforth) the particle is made to skip several data points in between along the trajectory when it is far away from the asteroid, just to reduce the length of the animation. So because of this, the particle appears to be moving faster when it is away from the asteroid but this is not true. For the exact velocity of the particle, the reader should look at the velocity magnitude indicator within the animation itself.

The animation shows that the particle reverses its direction of motion twice in its entire course. To visualize how this is happening in 3D, look at Figure 7.9. The reason for this can be understood by looking at the direction of the perturbing acceleration, the gravitational acceleration vectors, and the combined effect of all accelerations acting on the particle. The direction of SRP and STBE are shown in Figure 7.11 and that of the net effect of the two is shown in Figure 7.12a. In the trajectory simulator, the gravity model (triaxial ellipsoid model) computes the acceleration in the rotating frame. We calculated the direction of the gravitational acceleration in inertial frame in post-simulation analysis assuming a point-mass model by considering the fact that when the regolith is far-away from the asteroid, its gravity field would appear as that of a point-mass gravity source. The gravitational acceleration vectors are shown in Figure 7.12b. The net acceleration acting on the particle is then shown in Figure 7.13. All acceleration vectors are shown along those parts of the trajectory where the magnitude of SRP acceleration is of the same order of magnitude as that of the gravitational acceleration. However, the magnitude of the STBE acceleration is always 1.0 order of magnitude smaller than the gravitational acceleration for those very same points along the trajectory, but is still significant. We do not show the vectors for the entire trajectory for two reasons; first, when close to the asteroid the direction of these vectors would reduce the clarity of the plot and, second, we want to discuss the effect of the perturbations when the particle is far away from the asteroid because then they are as significant as the gravitational force.

In Figure 7.10, the trajectory loops numbered 1 and 2 (XY plane), is where the particle's direction of motion gets reversed. If we look at Figure 7.11a, we see that the direction of the SRP vector is consistent with how the particle changes its direction of motion. This, however, does not mean that the SRP is the sole actor responsible for how the particle's motion eventually turns out to be (and we will see this in detail shortly). The direction of STBE, as shown in Figure 7.11b, however, does not directly tell us on how the particle's motion would change as it progresses through its trajectory. STBE is always an order of magnitude smaller than SRP for the points shown in the two plots and its direction is not consistent with how the particle changes its direction of motion but its contribution to the capture scenario is significant (we will see the effect of removing STBE shortly). The direction of the net perturbing acceleration, shown in Figure 7.12a, shows us exactly how and where the motion of the particle is directed. Especially when we look at trajectory loops 1 and 2 in Figure 7.10, we can see that the net perturbing vector is acting in the direction that is consistent with how the particle changes its orbital motion. Now looking at these plots that we just discussed, a question that arises is that - why did the particle remain in a temporary capture orbit, and for example not escape especially when the net perturbing acceleration was acting opposite to the direction of asteroid such as in trajectory loop number 3 in Figure 7.10? The answer to this is found by looking at the direction of gravitational attraction in Figure 7.12b and the total acceleration (i.e. the net effect of gravity and perturbations) acting on the particle in Figure 7.13. Although the gravitational acceleration has the same order of magnitude as that of the Solar perturbations when the particle is far away from the asteroid, we see that the net effect of the two is towards the asteroid and hence prevents the particle from escaping.

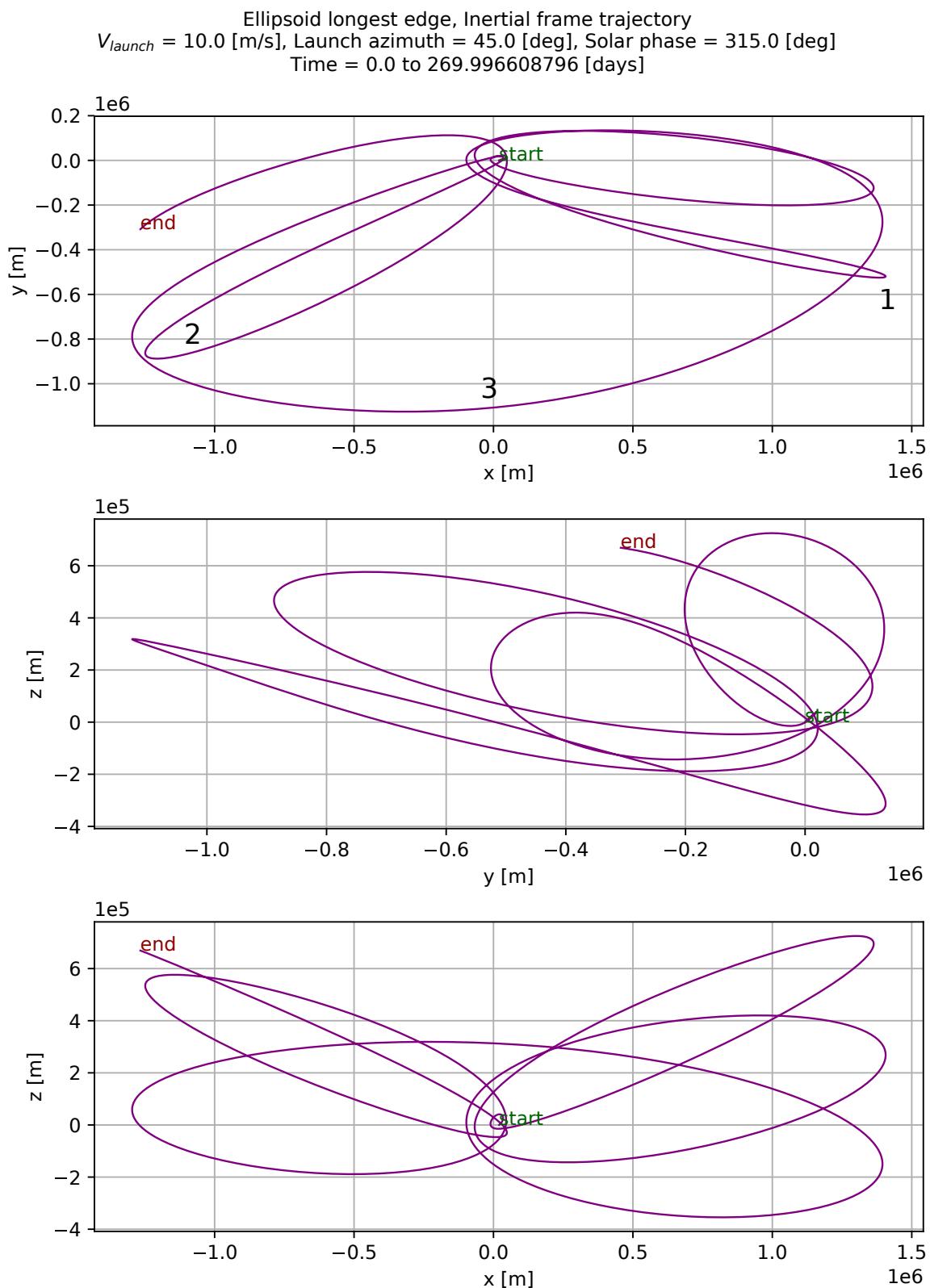
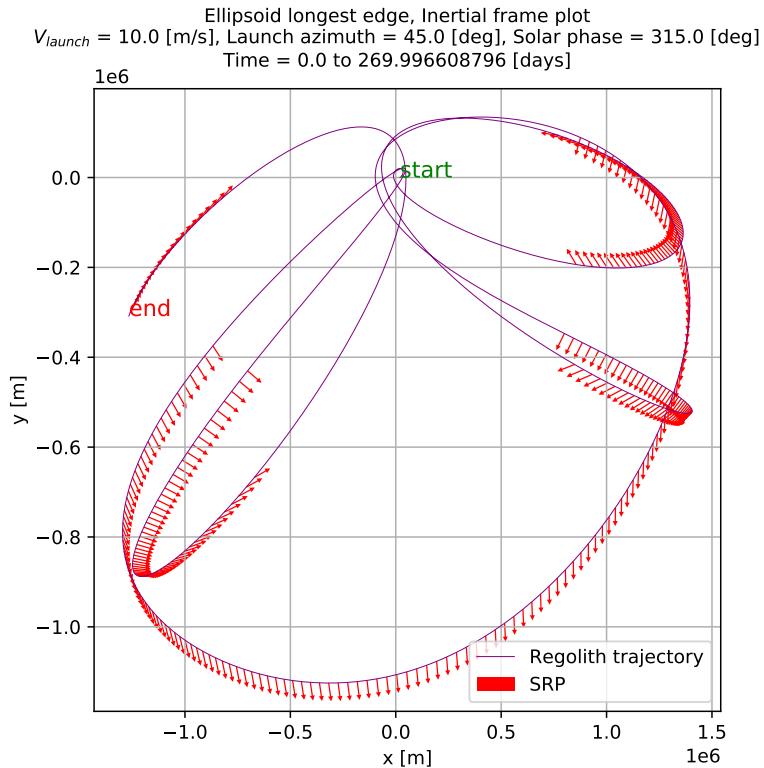
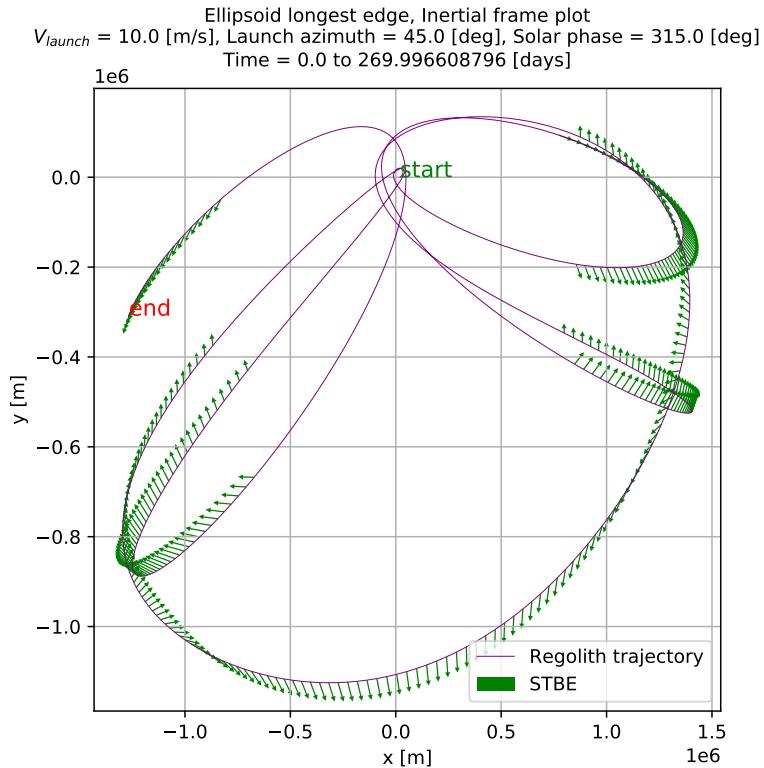


Figure 7.10: 2D inertial frame trajectory of capture regolith for case number 5 in Table 7.2. Particle code LoGSP-1.

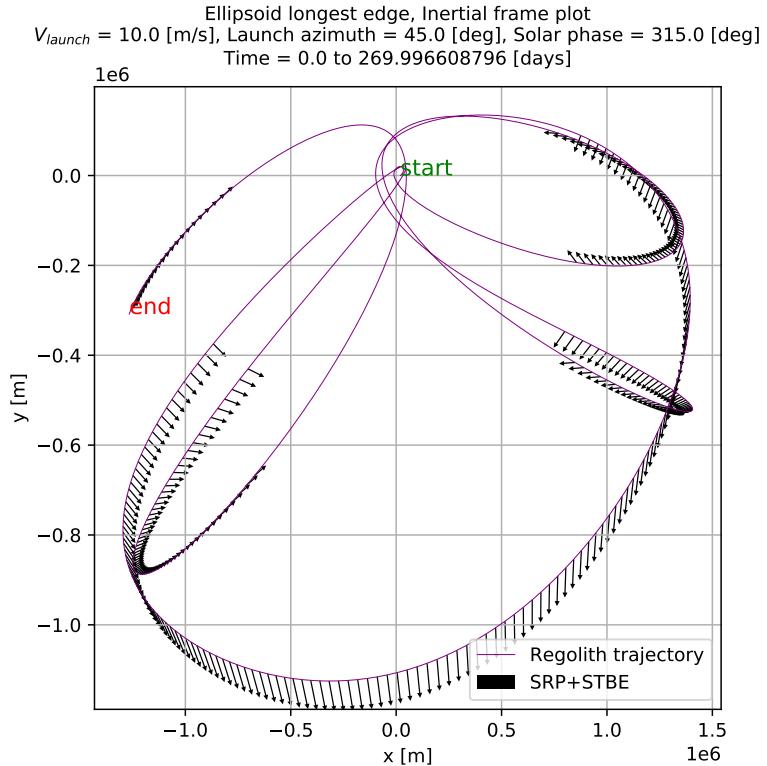


(a)

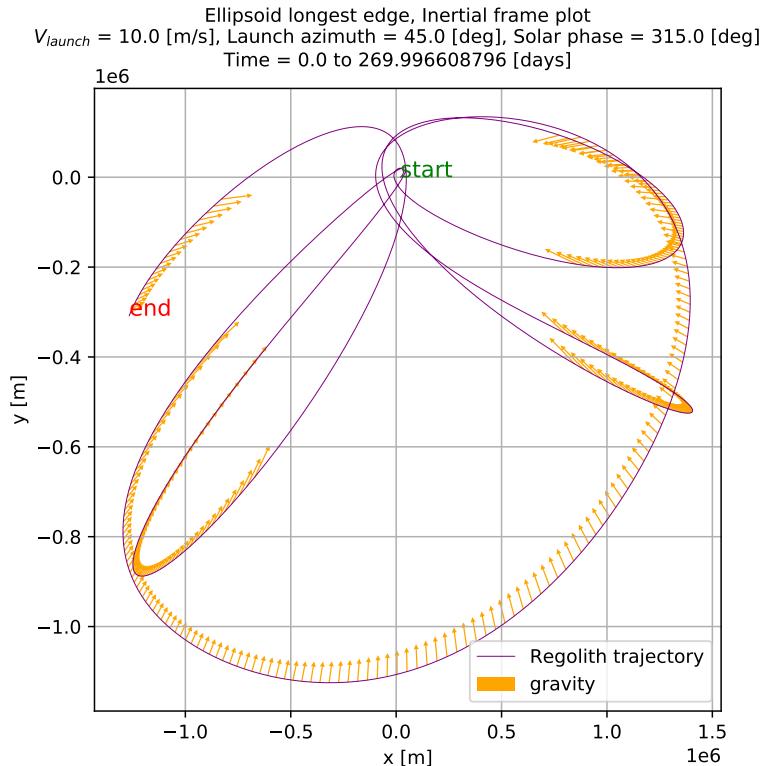


(b)

Figure 7.11: 2D trajectory of capture regolith for case number 5 in Table 7.2 with direction of SRP and STBE perturbation vectors. Note that the vectors are shown only for those parts of the trajectory where the SRP magnitude is of the same order as that of the asteroid's gravitational acceleration. For those very same points along the trajectory, the magnitude of the STBE is always 1.0 order of magnitude smaller than the gravitational acceleration. Particle code LoGSP-1.



(a)



(b)

Figure 7.12: 2D trajectory of capture regolith for case number 5 in Table 7.2 with direction of the sum total of SRP and STBE perturbation vectors, and the direction of the gravitational acceleration vector for the same data points. Note that the vectors are shown only for those parts of the trajectory where the SRP magnitude is of the same order as that of the asteroid's gravitational acceleration. For those very same points along the trajectory, the magnitude of the STBE is always 1.0 order of magnitude smaller than the gravitational acceleration. Particle code LoGSP-1.

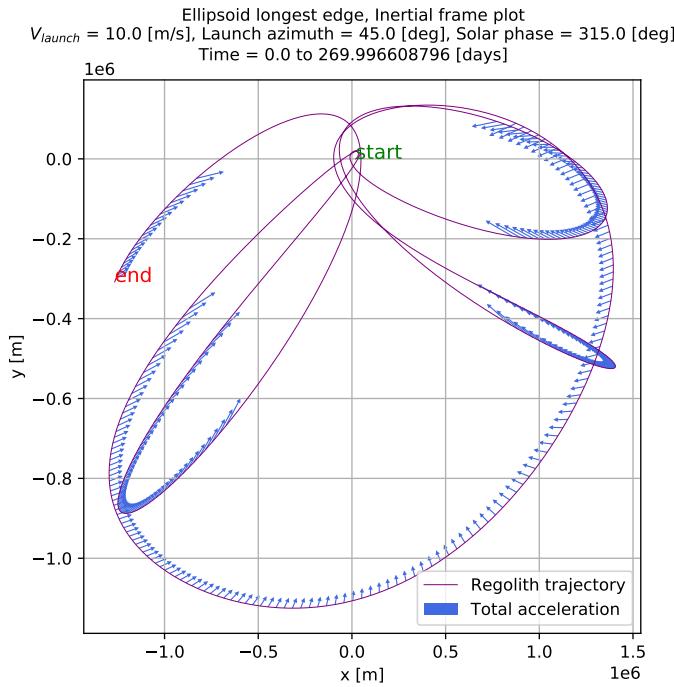


Figure 7.13: 2D trajectory of capture regolith for case number 5 in Table 7.2 with direction of the net acceleration vector.

Note that the vectors are shown only for those parts of the trajectory where the SRP magnitude is of the same order as that of the asteroid's gravitational acceleration. For those very same points along the trajectory, the magnitude of the STBE is always 1.0 order of magnitude smaller than the gravitational acceleration. Particle code LoGSP-1.

Both SRP and STBE together are necessary in getting the capture trajectory shown in Figure 7.10. If either one of them is removed from the simulation, for the same launch conditions and initial Solar phase angle, then the results are completely different and we do not get a capture orbit. Note that the definition of capture orbit in this context implies that the particle stays in an orbit around the asteroid for the complete duration of 270.0 [days], i.e., the maximum time for which the simulation is run.

EFFECTS OF OMITTING SOLAR THIRD BODY EFFECT (STBE)

When only STBE is removed, then we get a trajectory where the particle eventually escapes the asteroid. This is shown in Figure 7.14. The trajectory is completely different from the one in Figure 7.10, even though the only difference between the two simulations is the omission of STBE perturbation. Figure 7.15a shows the direction of perturbing acceleration due to SRP and the gravitational acceleration for those points along the trajectory where both have the same order of magnitude. The direction for the net acceleration acting on the particle is shown in Figure 7.15b. The trajectory of the particle starts out the same way in both Figure 7.14 and Figure 7.10 however due to the lack of STBE perturbation, the trajectories soon start to differ from each other. Upon comparing Figure 7.13 and Figure 7.15b we can infer that the trajectories differ because the with the lack of STBE the direction of the net acceleration vector differs for the two trajectories which eventually directs how the particle motion would progress. The particle trajectory in Figure 7.14 eventually leads to an escape scenario. Now if we look at Figure 7.15a, towards the end of the trajectory, the direction of the gravitational vector gradually changes and starts to point along the instantaneous tangent to the trajectory, all the while with the SRP vector pointing away from the asteroid. The net effect of this situation can be seen in Figure 7.15b; we see that the net acceleration vector starts pointing away from the asteroid towards the end segment of the trajectory and thus this is when the particle escapes.

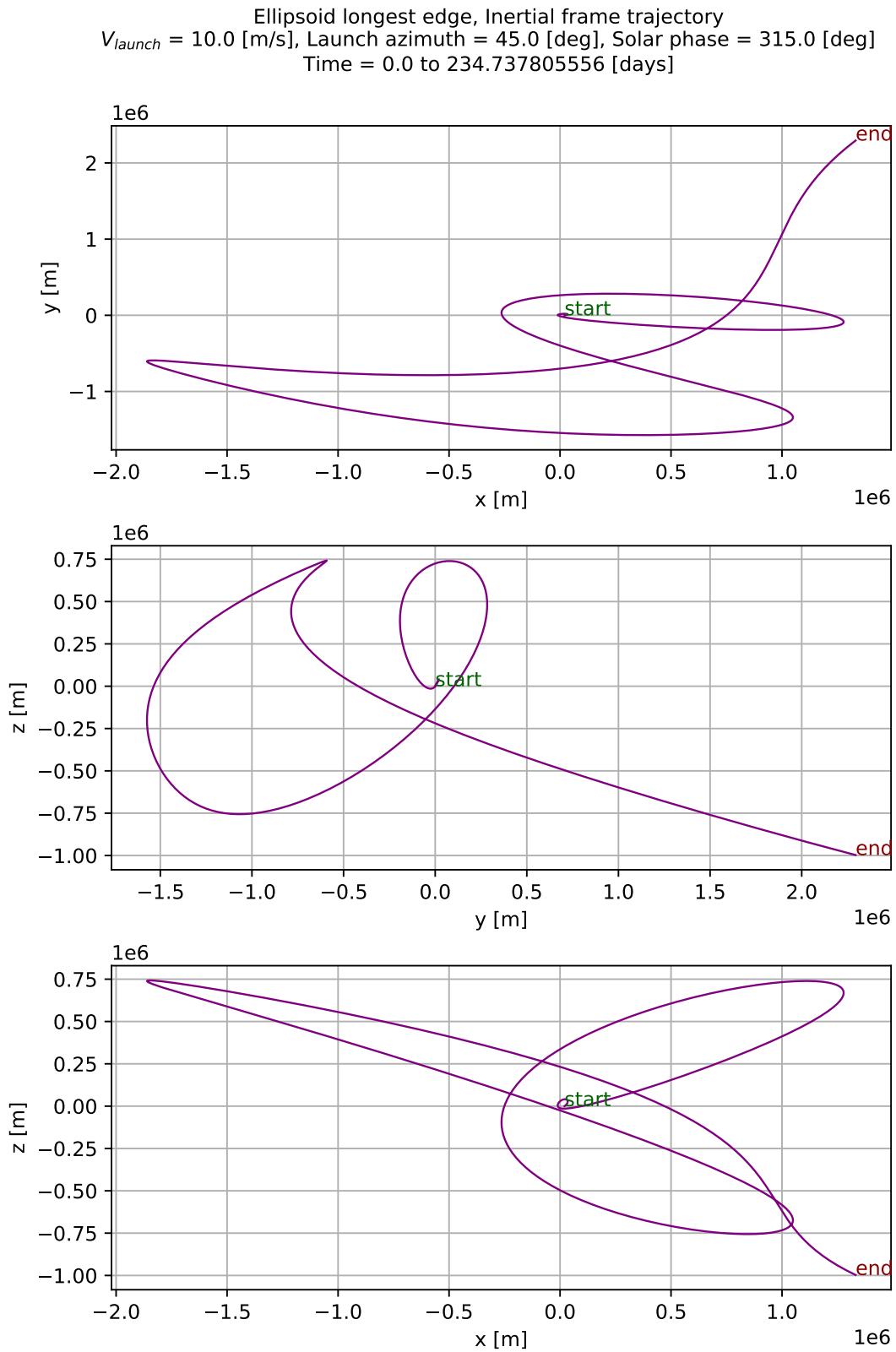
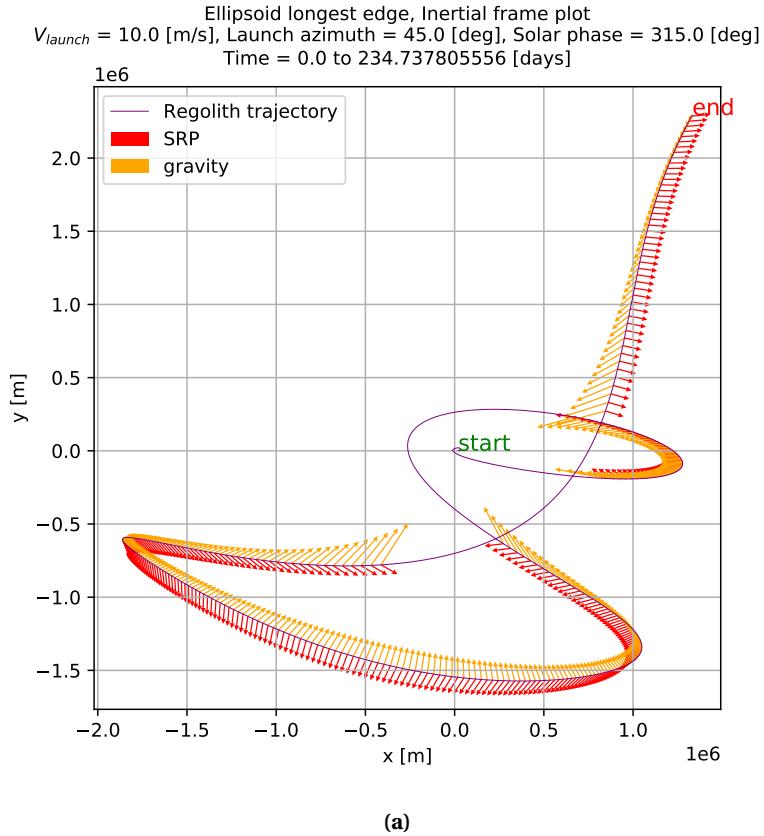
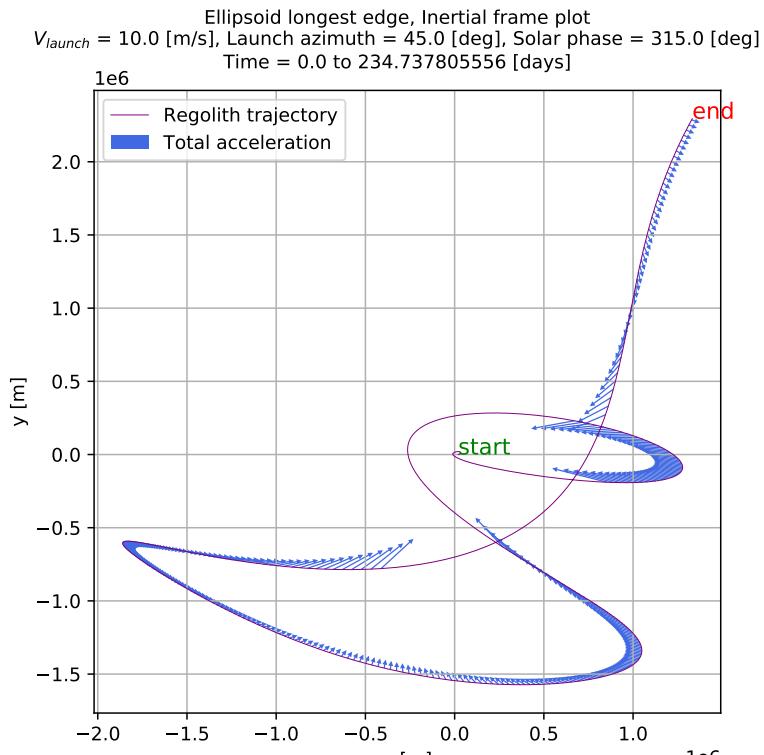


Figure 7.14: 2D trajectory of particle for same initial conditions as that of capture case 5 in Table 7.2 except that only SRP was included in this simulation. Particle code LoGSP-1.



(a)



(b)

Figure 7.15: Inertial frame XY plane trajectory for same launch conditions as that of capture case 5 in Table 7.2: (a) showing direction of SRP acceleration and gravitational acceleration & (b) showing direction of the net acceleration acting on the particle. Vectors are shown only for those parts of trajectory where acceleration due to SRP and gravity have the same order of magnitude. Note that STBE perturbation was not part of the simulation here. Particle code LoGSP-1.

EFFECTS OF OMITTING SOLAR RADIATION PRESSURE (SRP)

When we keep STBE but remove SRP from our simulations, then the trajectory again leads to an escape situation, only this time it's much faster. The 2D inertial frame trajectory is shown in Figure 7.17. The trajectory shows similarity only for a brief moment immediately after launch (notice the small loop after 'start') with the capture case in Figure 7.10 but soon after the particle is on a trajectory that never comes back around the asteroid. The reason for this is clear and simple if one looks at the direction of acceleration due to gravity and STBE in Figure 7.18a and their net effect in Figure 7.18b. Initially, from the point when we show these vectors, we know that the magnitude of STBE acceleration is 1.0 order of magnitude smaller than the gravitational acceleration (see Figure 7.19) and even then the direction of the net acceleration vector is such that the trajectory can not loop around the asteroid. The STBE magnitude increases soon enough to the same order as that of gravitational acceleration and the net acceleration vector direction never points towards the asteroid which eventually causes the particle to escape. However, the point where the magnitude curves of STBE and gravitational acceleration cross is not the point where the escape occurs as is evident from the plot for total energy and eccentricity in Figure 7.16.

From this analysis, we can say that effect of removing SRP from simulations had a much drastic effect than removing just the STBE. Both cases lead to an escape situation and the combined effect of both the perturbations leads to a capture orbit, for the same launch conditions and initial Solar phase angle. The behavior of the trajectory, in all cases, can be easily understood by looking at the direction of the net acceleration vector, especially when the particle is far away from the asteroid because it tells us exactly, how by adding perturbations, the motion of the particle is affected and not just in terms of its final fate but even in terms of changing its orbital direction.

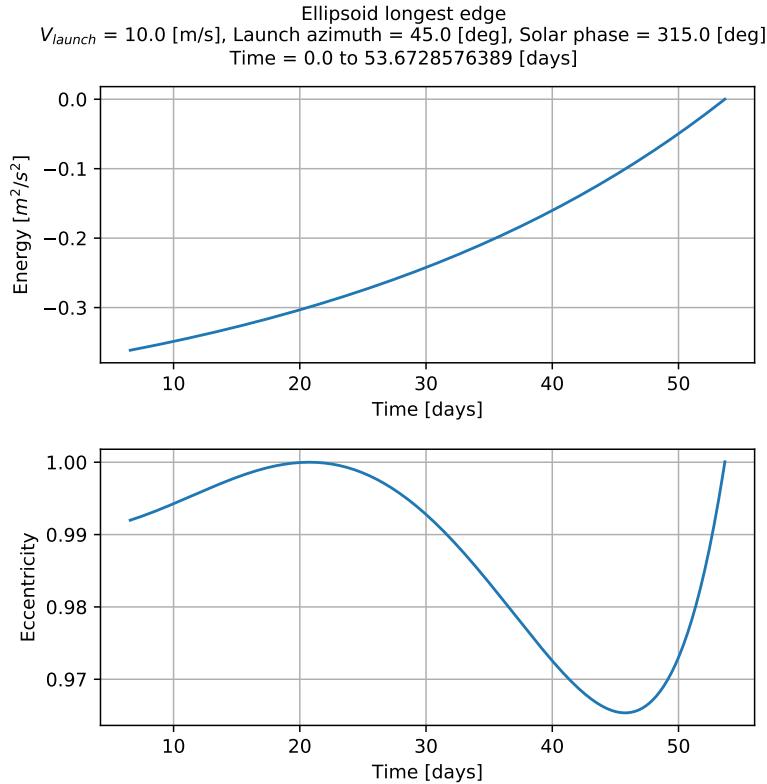


Figure 7.16: Evolution of total energy of the particle and its orbital eccentricity. Particle has the same initial conditions as that of capture case 5 in Table 7.2 except that only STBE was included in this simulation. The range of data points plotted is the same as that in Figure 7.18. Particle code LoGSP-1.

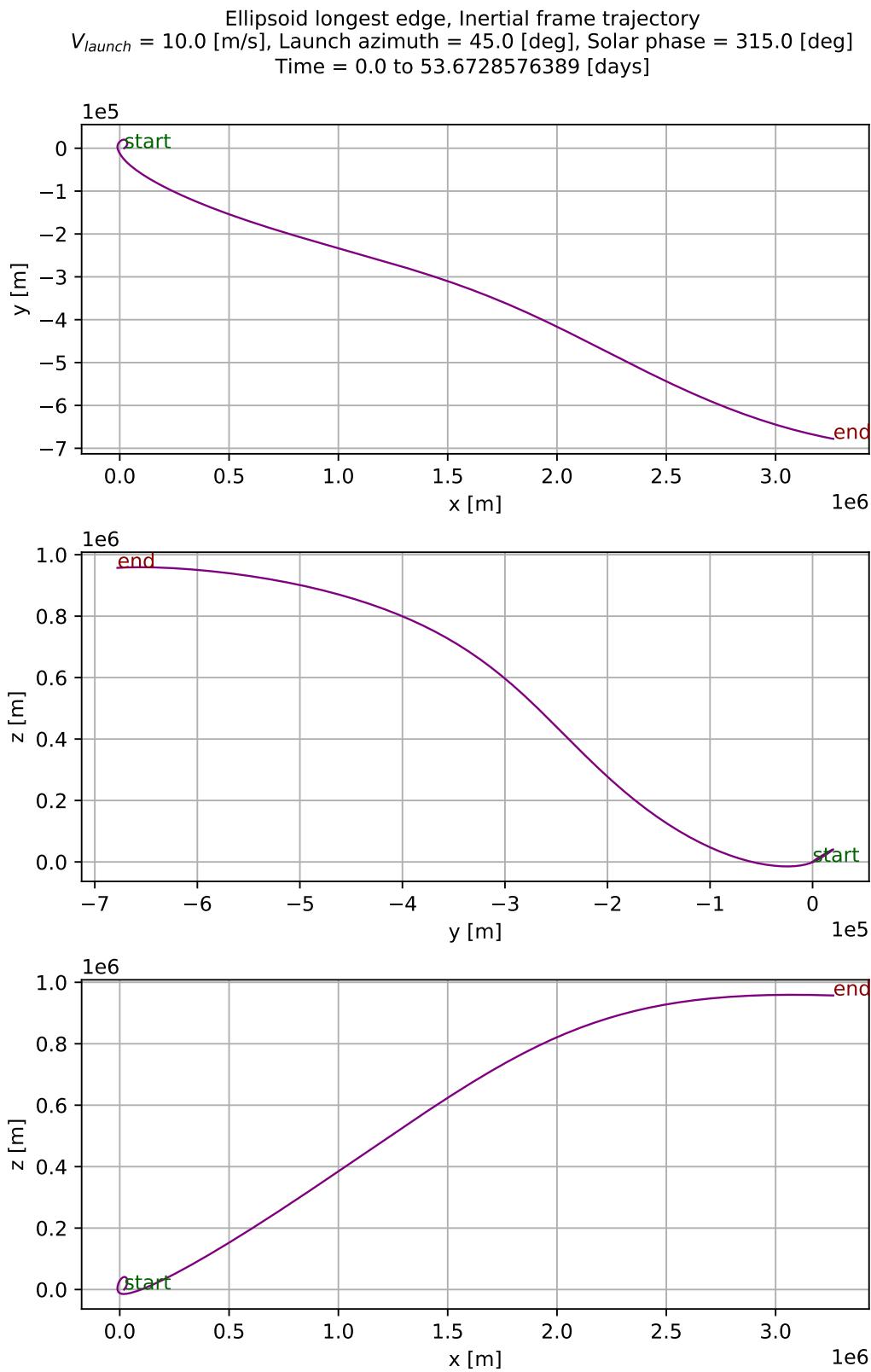
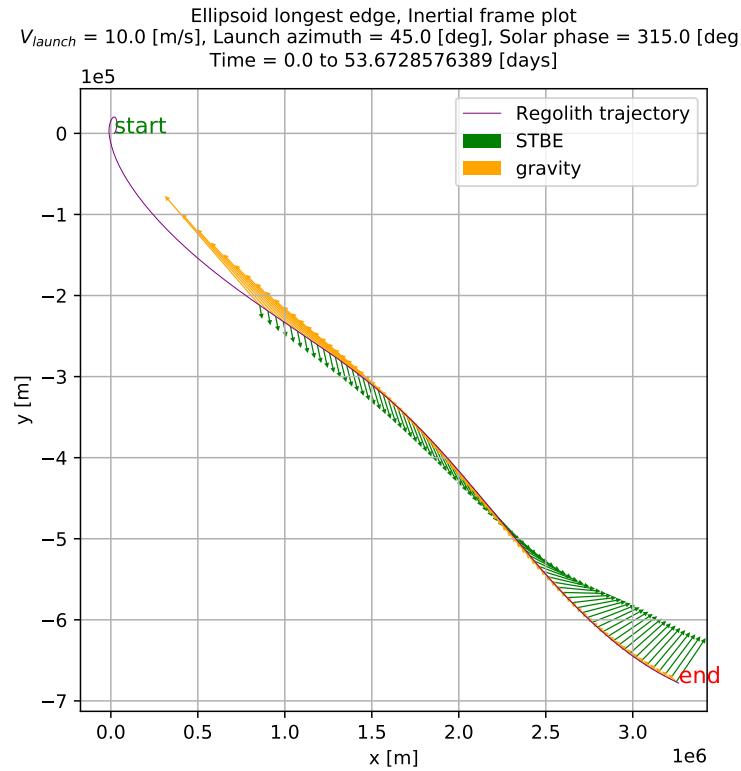
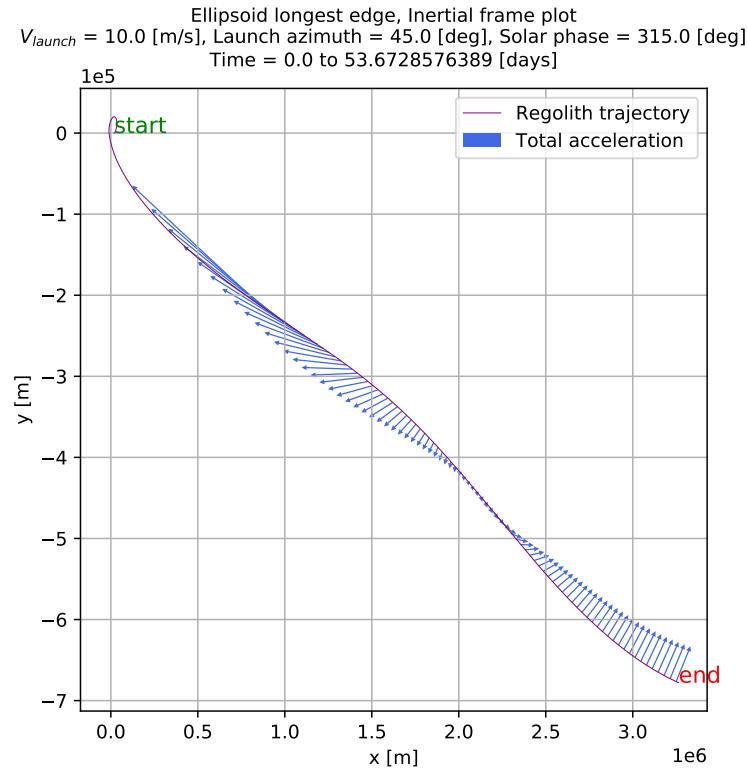


Figure 7.17: 2D trajectory of particle for same initial conditions as that of capture case 5 in Table 7.2 except that only STBE was included in this simulation. Particle code LoGSP-1.



(a)



(b)

Figure 7.18: Inertial frame XY plane trajectory for same launch conditions as that of capture case 5 in Table 7.2: (a) showing direction of STBE acceleration and gravitational acceleration & (b) showing direction of the net acceleration acting on the particle. Vectors are shown only for those parts of trajectory where acceleration due to STBE is equal to gravitational acceleration or smaller than it by 1.0 order of magnitude. Note that SRP perturbation was not part of the simulation here. Particle code LoGSP-1.

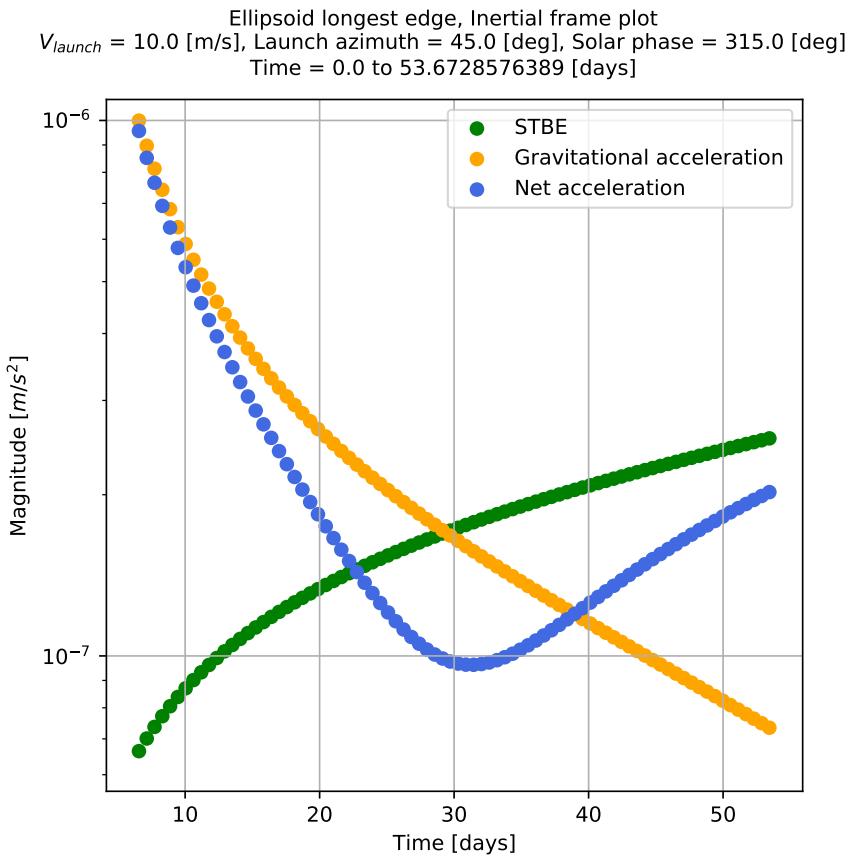
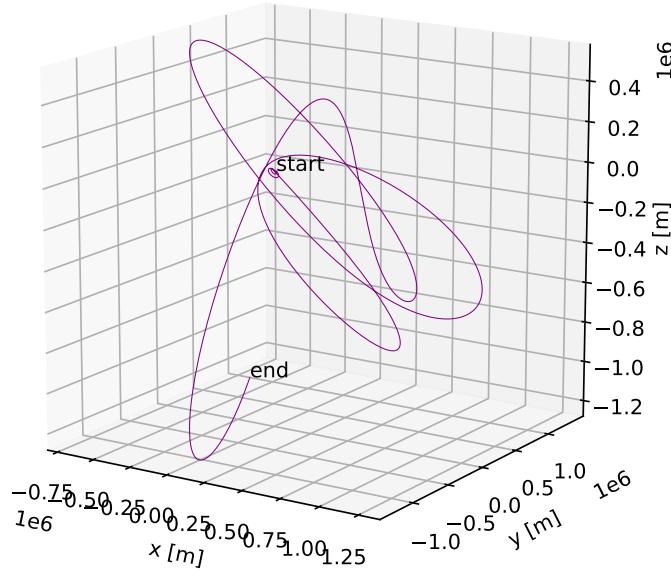


Figure 7.19: Magnitudes of acceleration due to gravity, STBE and the net effect of the two for the corresponding vectors as shown in Figure 7.18. Particle code LoGSP-1.

CAPTURE TRAJECTORY: ANOTHER EXAMPLE

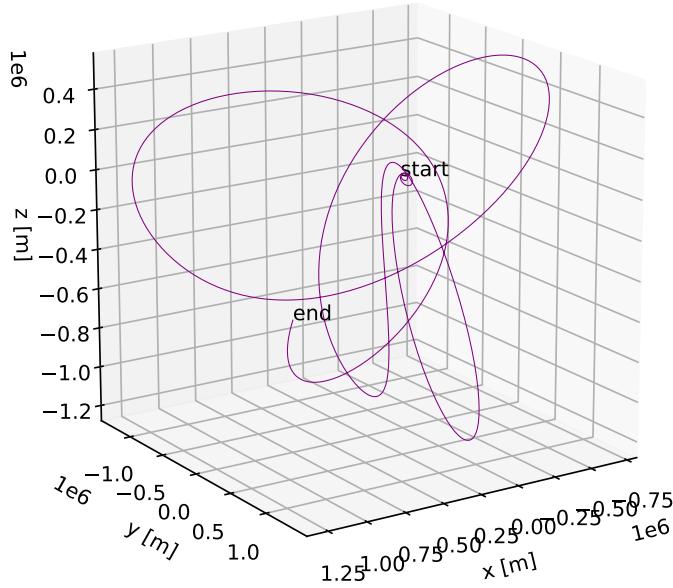
Figure 7.20 shows the 3D trajectory for completely different launch conditions (see capture case 8 in Table 7.2). The 3D trajectory as viewed from the asteroid centric body fixed frame is shown in Figure C.33. The 2D trajectory projections for the same, in inertial and body fixed frames, are shown in Figure 7.21 and Figure C.34 respectively. Just like in the previous case, we see from the animation (see Figure 7.25) and the 3D trajectory for current launch conditions, that the particle direction of motion is reversed twice in its course. These two locations are marked by numbers 1 and 2 in Figure 7.21. At location number 1 we see that the motion changes from anti-clockwise to clockwise direction in the XY plane. The case for location number 2 is exactly the opposite. If we look at Figure 7.23a, the change in direction of motion is consistent with the direction in which the net perturbing force is acting. Ultimately, when we look at the net acceleration acting on the particle in Figure 7.24, we can understand how exactly the particle would orbit around the asteroid. The net force, gravitational and perturbations combined, act in a direction such that the particle is forced to change its orbital motion direction at the two locations previously explained. The acceleration vectors in Figures 7.22 to 7.24 are plotted for points along the trajectory where the magnitude of acceleration due to SRP is of the same order of magnitude as the gravitational acceleration. Again, the magnitude of STBE is 1.0 order of magnitude smaller than the gravitational acceleration for the same data points along the trajectory.

Ellipsoid longest edge, Inertial frame trajectory
 $V_{\text{launch}} = 8.0$ [m/s], Launch azimuth = 165.0 [deg], Solar phase = 45.0 [deg]
Time = 0.0 to 270.0 [days]



(a)

Ellipsoid longest edge, Inertial frame trajectory
 $V_{\text{launch}} = 8.0$ [m/s], Launch azimuth = 165.0 [deg], Solar phase = 45.0 [deg]
Time = 0.0 to 270.0 [days]



(b)

Figure 7.20: 3D inertial frame trajectory of capture regolith for case number 8 in Table 7.2 from two different viewing angles. Particle code LoGSP-1.

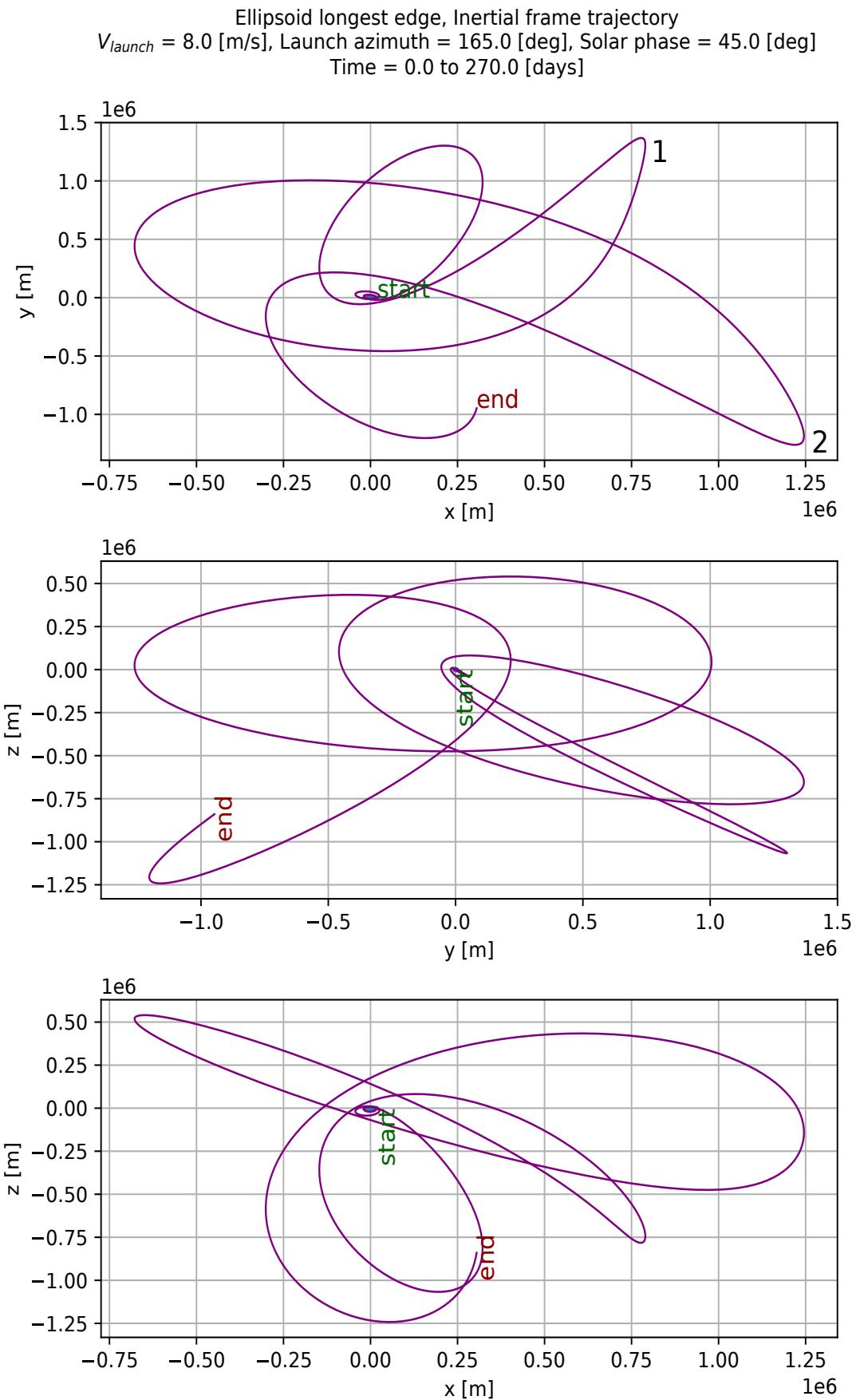
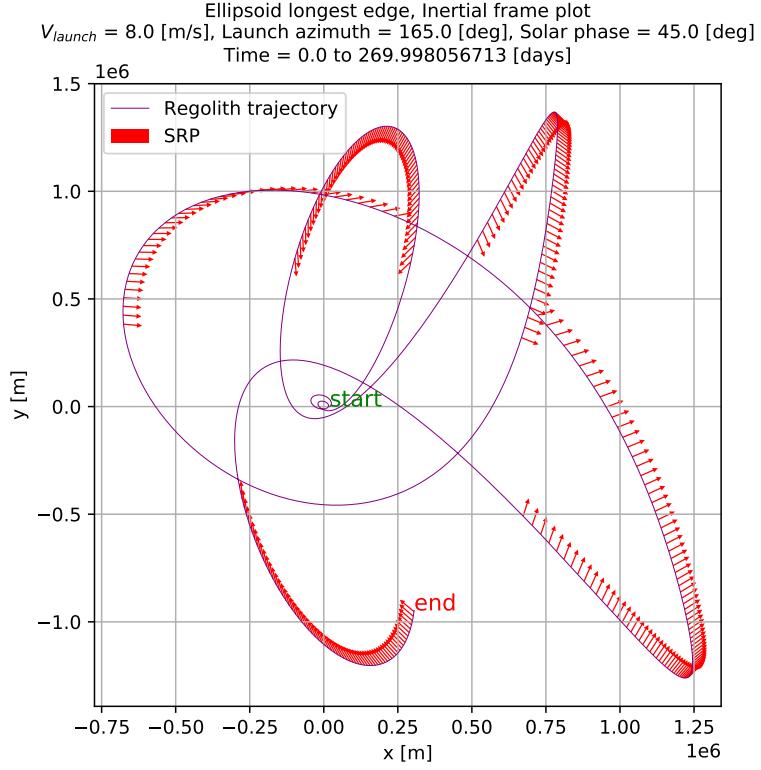
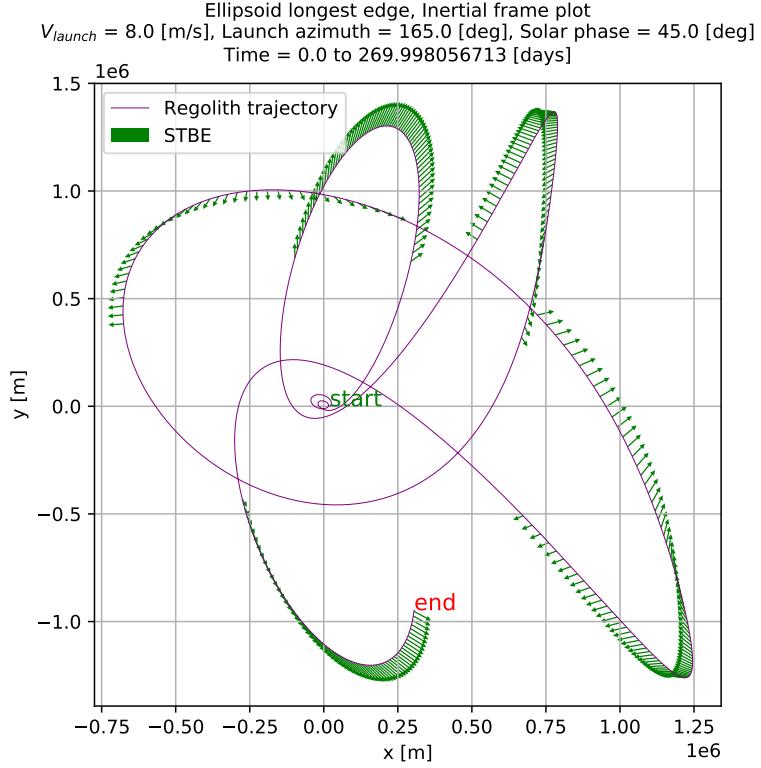


Figure 7.21: 2D inertial frame trajectory of capture regolith for case number 8 in Table 7.2. Particle code LoGSP-1.



(a)



(b)

Figure 7.22: 2D trajectory of capture regolith for case number 8 in Table 7.2 with direction of SRP and STBE perturbation vectors. Note that the vectors are shown only for those parts of the trajectory where the SRP magnitude is of the same order as that of the asteroid's gravitational acceleration. For those very same points along the trajectory, the magnitude of the STBE is always 1.0 order of magnitude smaller than the gravitational acceleration. Particle code LoGSP-1.

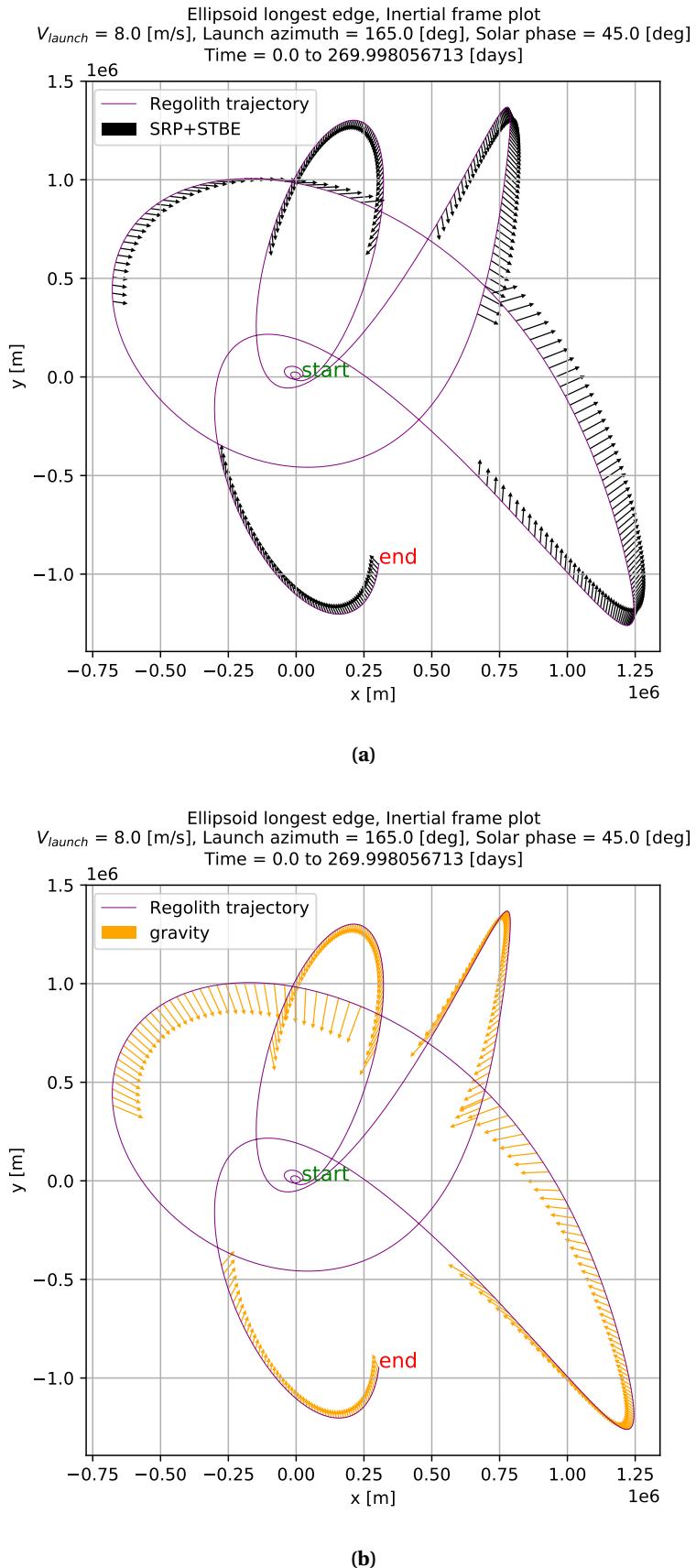


Figure 7.23: 2D trajectory of capture regolith for case number 8 in Table 7.2 with direction of the sum total of SRP and STBE perturbation vectors, and the direction of the gravitational acceleration vector for the same data points. Note that the vectors are shown only for those parts of the trajectory where the SRP magnitude is of the same order as that of the asteroid's gravitational acceleration. For those very same points along the trajectory, the magnitude of the STBE is always 1.0 order of magnitude smaller than the gravitational acceleration. Particle code LoGSP-1.

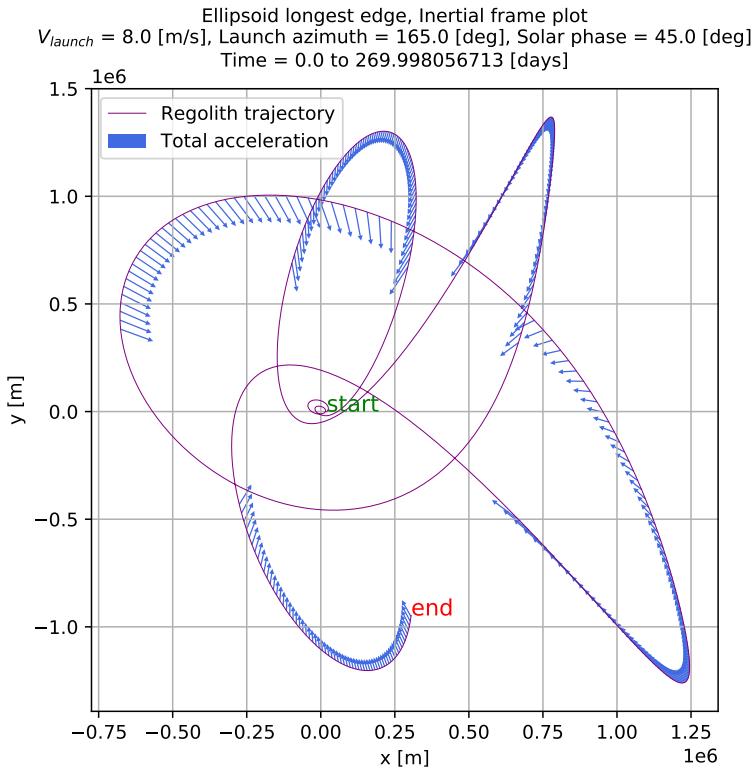


Figure 7.24: 2D trajectory of capture regolith for case number 8 in Table 7.2 with direction of the net acceleration vector. Note that the vectors are shown only for those parts of the trajectory where the SRP magnitude is of the same order as that of the asteroid's gravitational acceleration. For those very same points along the trajectory, the magnitude of the STBE is always 1.0 order of magnitude smaller than the gravitational acceleration. Particle code LoGSP-1.



Figure 7.25: 2D trajectory animation (XY Plane) of capture regolith for case number 8 in Table 7.2. Particle code LoGSP-1. Scan the QR code to view the animation or use the following web-link: <https://youtu.be/CceYR1NvAiM>

HOW ARE CAPTURE ORBITS FORMED

We saw in the analysis of capture case number 5 for particle LoGSP-1 that both SRP and STBE were necessary for getting that specific capture trajectory and removal of either of the perturbations resulted in a different final fate for the same particle. The next analysis that we present now, will tell us about how a capture scenario occurs, relative to a situation when all perturbations are removed, for the same initial launch conditions in both cases. We do the analysis for capture case number 8 from Table 7.2. Figure 7.27a shows two different trajectories for the particle launched with the same initial conditions. The one shown in dotted line is for the case when Solar perturbations were omitted from the simulation, which eventually results in the particle escaping the asteroid after 1.4 [days]. The one in the solid line shows the capture trajectory (actually a section of the entire capture trajectory as seen in Figure 7.21) when Solar perturbations were included in the simulation. Note that we show the perturbed trajectory (capture case) for the same amount of time (1.4 [days] instead

of 270.0 [days]) as taken by the unperturbed trajectory (escape case) to be able to do a one-to-one comparison. The arrows plotted along this trajectory indicate the direction of the net perturbing acceleration due to SRP and STBE. Figure 7.26 directs to an animation for both the unperturbed and perturbed trajectory.

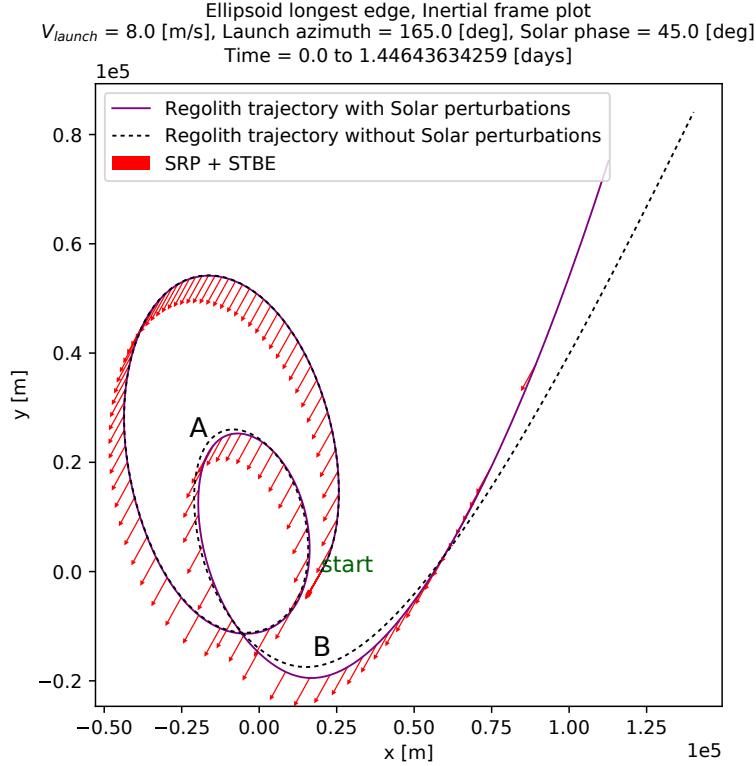


Figure 7.26: 2D trajectory animation (XY Plane) of capture regolith for case number 8 in Table 7.2, compared with that of its unperturbed counterpart. Particle code LoGSP-1. Scan the QR code to view the animation or use the following web-link: <https://youtu.be/CdFKKR3UDJ0>

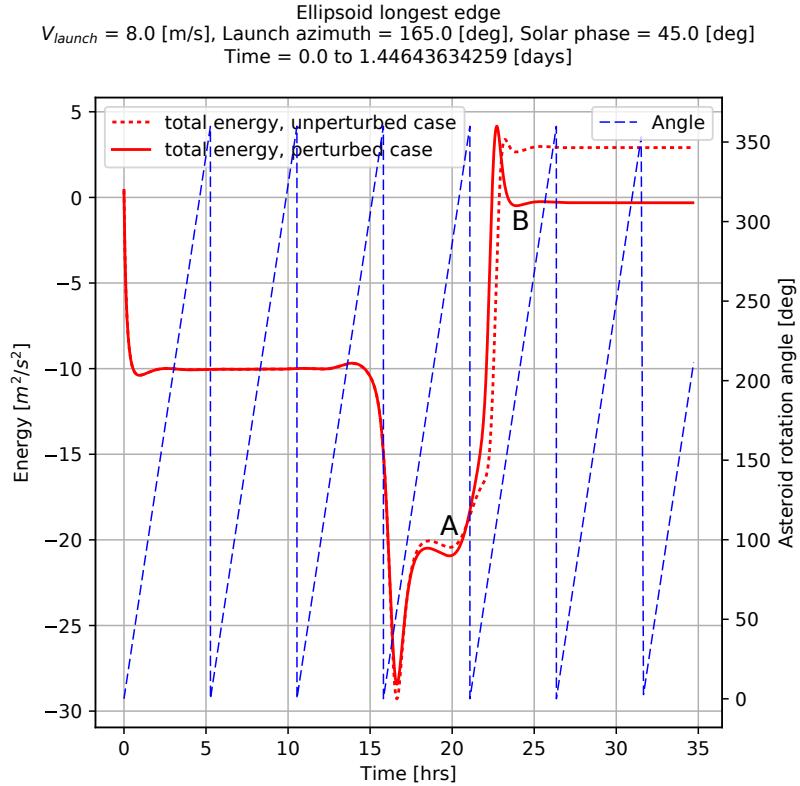
From the animation we can see that even as the particle has just been lofted from the surface of the asteroid, there are very subtle and minute differences in the range to the particle and its velocity, between the perturbed and unperturbed trajectory. The first visible difference between the two trajectories becomes noticeable at point "A" in Figure 7.27a. It is easy to deduce the change in the perturbed trajectory from the direction of the net perturbing acceleration up until this point. The same point "A" is also marked in Figure 7.27b. It is from this point that we can see noticeable difference between the two trajectories as well as in their corresponding energies. A snippet from the trajectory animation, corresponding to the point "A", is shown in Figure 7.31a which highlights the differences in range and velocity of the particles in the two trajectories. Note that in Figure 7.31a, the difference in the velocity between the perturbed and unperturbed cases is relatively small, compared to almost 1 [km] of a difference in range of the particles. The latter is significant since the particles have dimensions in the order of [cm]. From point "A" onwards these differences continue to grow and only get larger as the trajectory proceeds.

Similarly, at point "B" in Figure 7.27a, we see a much larger difference in the two trajectories. In Figure 7.27b, we see that around point "B" both trajectories have a positive energy which quickly comes down to a negative value for the perturbed trajectory, hence keeping it bounded which results in a capture scenario. However, this does not happen for the unperturbed trajectory, leading to an escape scenario. The difference in the state of the two particles at point "B" are relatively larger and can be seen in Figure 7.31b. The differences in the two trajectories, computed in the asteroid centric rotating frame, is shown in Figure 7.28. The plot on the bottom shows the trajectory for 1.4 [days] (i.e. until escape for the unperturbed trajectory) as viewed in the rotating frame, and the plot on the right zooms into a small part of this trajectory to show how Solar perturbations are responsible for changing the course of the particle. It is seen with a bit more clarity on how the net perturbation vector pulls the trajectory away from the trace of the unperturbed one.

So what we are seeing here is, that due to the inclusion of perturbations from the Sun, the motion of the particle changed from its unperturbed counterpart. This change was not drastic in terms of the initial shape of the trajectory as seen in Figure 7.27a. But the change was just enough for the particle to have a different phase with respect to the asteroid, relative to the unperturbed trajectory as seen in Figure 7.31. By phase, we refer to the location of the particle with respect to a given rotational state of the asteroid. So if two particles are at different locations, at any given epoch and for the same rotational state of the asteroid, they will have different magnitudes of forces acting on them which would ultimately lead to different final outcomes.



(a)



(b)

Figure 7.27: Comparative analysis of capture case 8 in Table 7.2 with a particle trajectory where the initial conditions are same as the former but the simulation was done without Solar perturbations. Figure 7.27a compares the XY plane trajectory & Figure 7.27b compares their total energy. Particle code LoGSP-1.

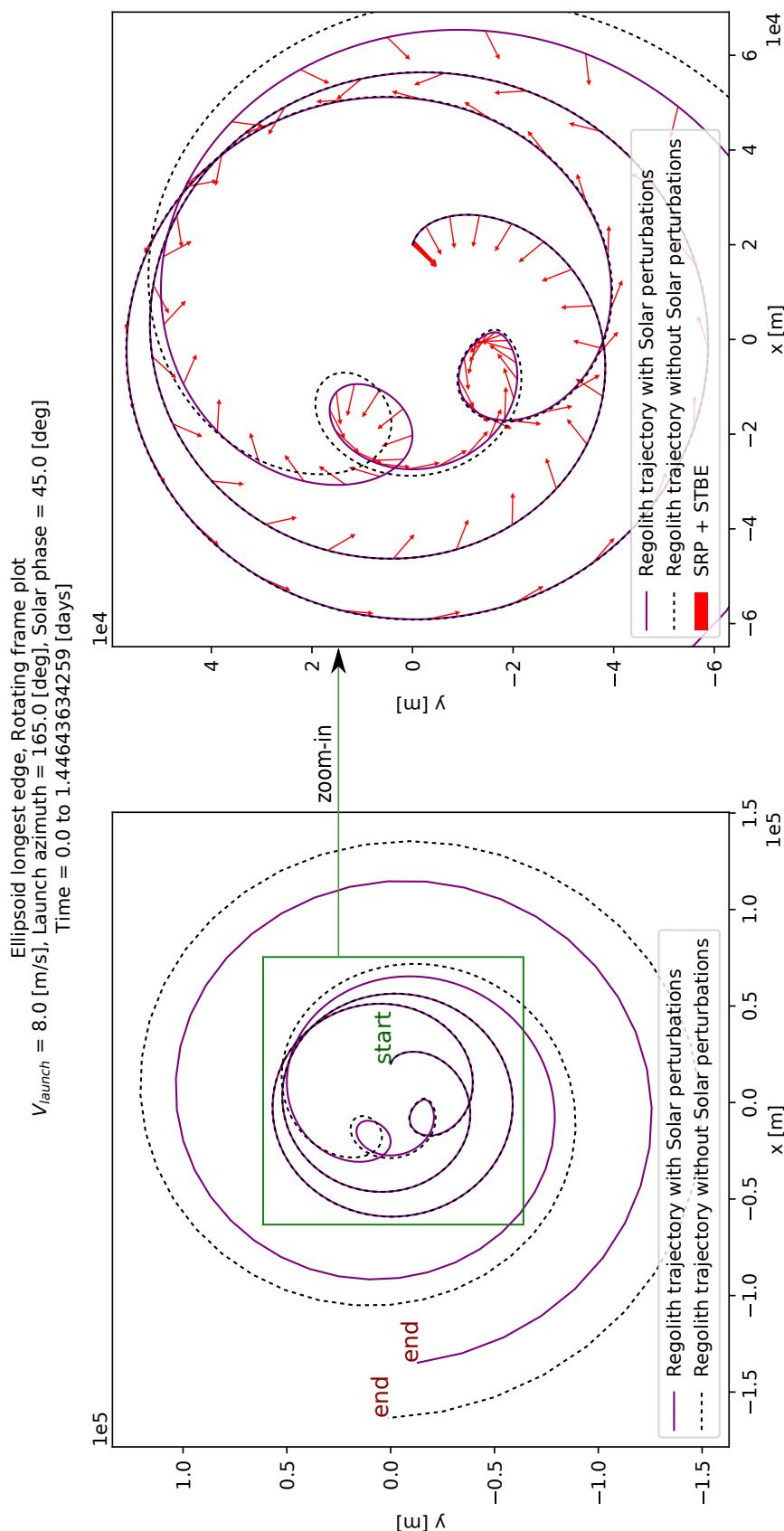


Figure 7.28: Rotating frame 2D trajectory (XY plane) of capture regolith for case number 8 in Table 7.2 with direction of the net perturbation vector, compared with the trajectory of a particle launched with the same initial conditions but in absence of Solar perturbations. Particle code LoGSP-1.

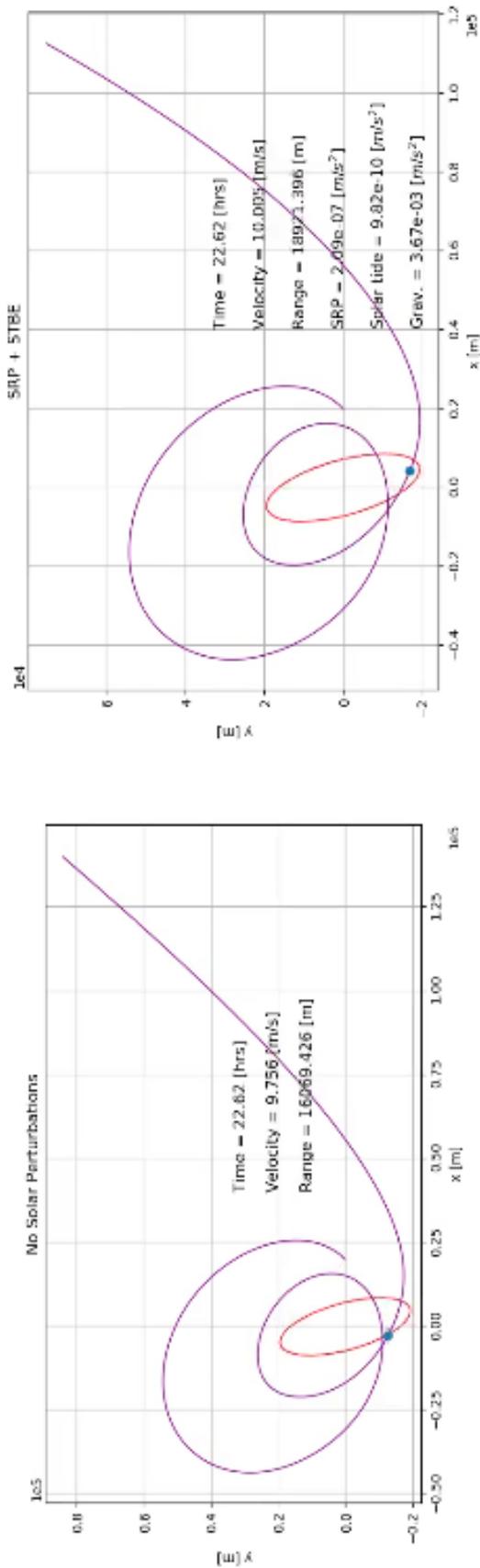


Figure 7.29: Snapshot from animation of the perturbed trajectory of capture case 8 in Table 7.2 compared with that of its unperturbed counterpart. The unperturbed trajectory is still being accelerated at the given instant however the particle in the perturbed trajectory is being decelerated. Particle code LoGSP-1.

If we look at the trajectory animation in Figure 7.26, one would notice that at around point "B", the particle in the unperturbed trajectory is being accelerated by the gravitational pull of the asteroid while the particle in the perturbed trajectory is being slowed down. A snapshot of this scenario from the animation is shown in Figure 7.29. Although this situation does not happen for extended periods of time, but only while approaching point "B", we see that the particle in the unperturbed trajectory has relatively higher velocity while moving forth of point "B" and leaving the vicinity of the asteroid, relative to the particle in the perturbed trajectory. The latter thus stays in a capture orbit while the former has enough velocity to escape. A plot for this is shown in Figure 7.30.

Note that in the capture case just discussed, the magnitude of the perturbing accelerations is much smaller than the gravitational acceleration. The effect of the perturbations on the particle's trajectory is not instantaneous and we can see that in the initial part of the trajectories up until point "A" in Figure 7.27a. Until this point, acceleration due to gravity is in the order of 10^{-4} , while accelerations due to SRP and STBE are in the orders of 10^{-7} and 10^{-9} respectively. Although the perturbing magnitudes are small, but the particles in question are extremely small as well and so over time, the perturbing accelerations add up, leading to a significant change in the trajectory from the unperturbed one.

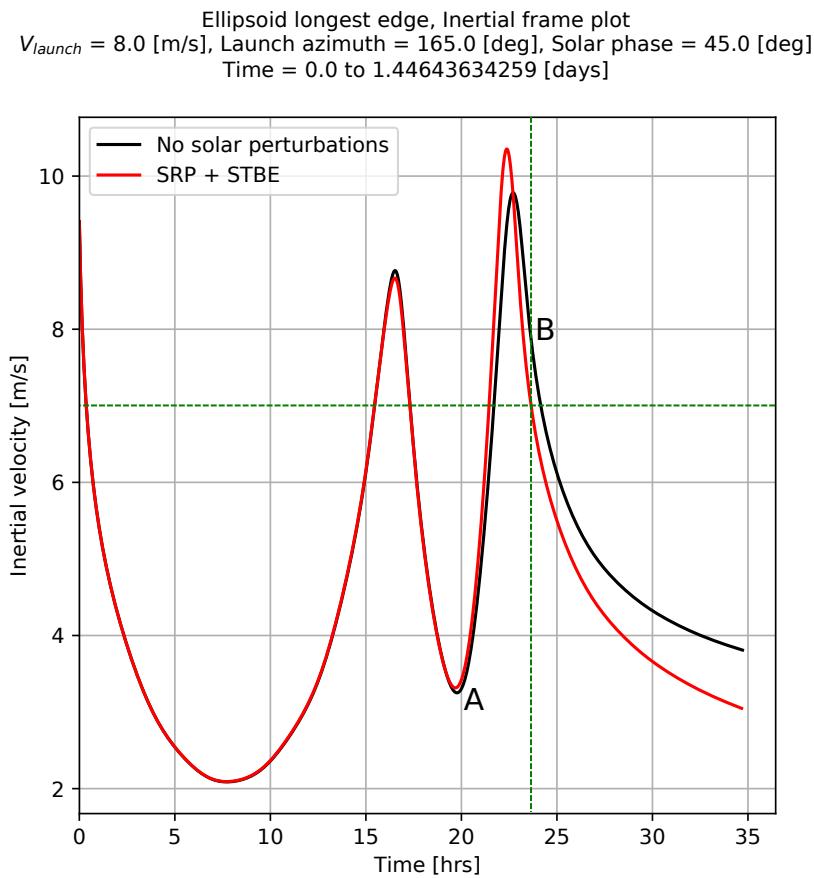


Figure 7.30: Inertial velocity of the perturbed trajectory of capture case 8 in Table 7.2 compared with that of its unperturbed counterpart. The trajectories are shown for the time it takes for the particle in the unperturbed trajectory to escape. Particle code LoGSP-1.

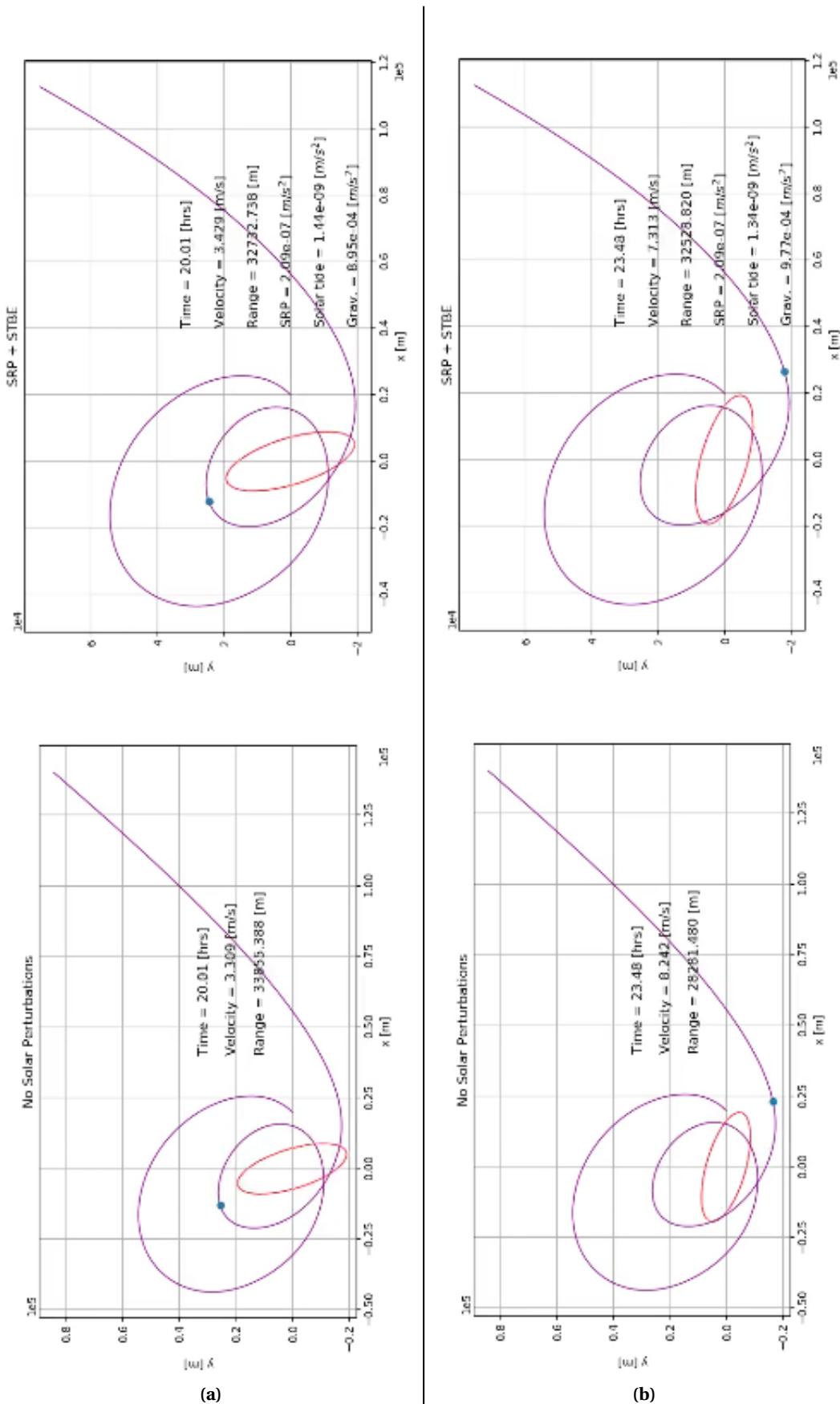


Figure 7.31: Animation snippets of the inertial frame 2D trajectory (XY plane) of capture regolith for case number 8 in Table 7.2. The bottom two plots are for the case when Solar perturbations were omitted from the simulation and the top two plots includes them. Note the differences in the range to the particle and its velocity for the same time stamp and rotational state of the asteroid. Particle code LoGSP-1.

We see a similar effect when we look at capture case number 5 from Table 7.2. The inertial frame trajectory, both perturbed and unperturbed, for it are shown in Figure 7.32. With Solar perturbations removed from the simulation, the initial conditions for this particle result in it getting launched on a highly elliptical orbit and eventually crashing onto the surface of the asteroid after 96 days. The particle, however, avoids this fate when Solar perturbations are included in the simulation. In Figure 7.32, it can be clearly seen that the direction of the perturbing acceleration due to SRP and STBE is consistent with how the trajectory departs from its unperturbed counterpart. The trajectories are shown only for the time it takes for the particle in the unperturbed trajectory to re-impact the surface of the asteroid. We show this case to highlight the effect perturbations have on a trajectory destined for re-impact, unlike the escape scenario discussed previously. We see drastic changes in the perturbed trajectory from the unperturbed one because when the particle is far away from the asteroid, the perturbing acceleration magnitude is of the same order as that of the gravitational acceleration.

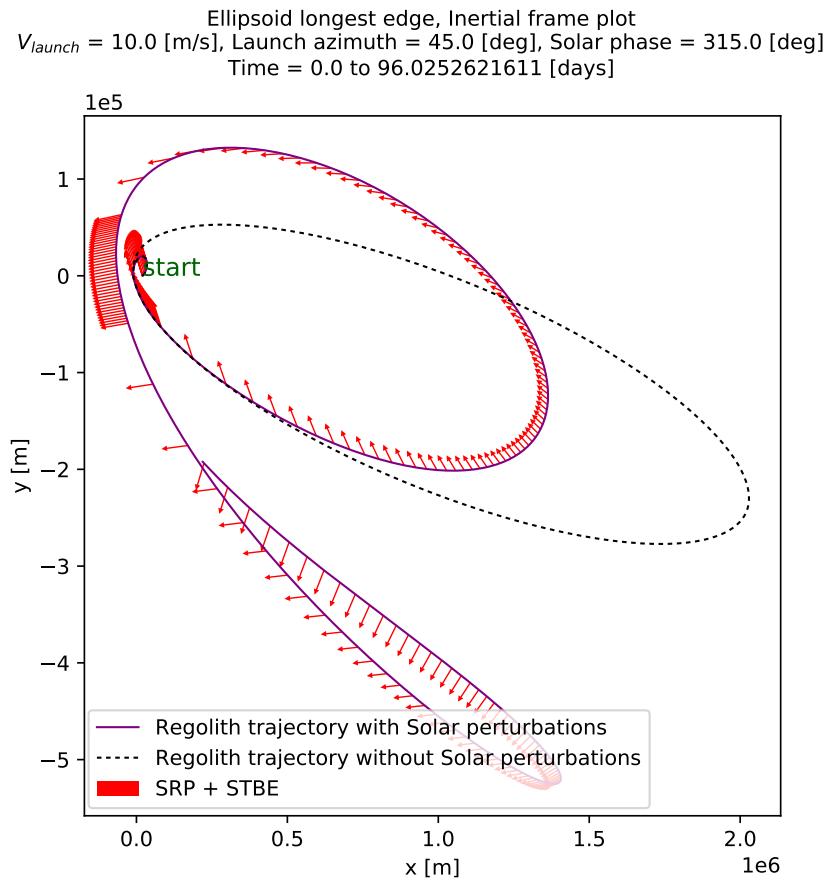


Figure 7.32: Inertial frame 2D trajectory (XY plane) of capture regolith for case number 5 in Table 7.2 with direction of SRP perturbation vector compared with the trajectory of a particle launched with the same initial conditions but in absence of Solar perturbations. Trajectories shown for as long as it takes the unperturbed trajectory Particle code LoGSP-1.

7.4 FINAL FATE BEHAVIOR OF DIFFERENT REGOLITH TYPES

For simulations accounting Solar perturbations, the discussion so far has been about how perturbations affect particle motion and specifically the capture scenario, relative to a particle in an unperturbed simulation. We did this detailed analysis for a single particle type only, namely LoGSP-1 from Table 7.1. We shall now look into the final fate behavior of all the regolith types mentioned in

Table 7.1 to understand how particle motion is affected for different densities and sizes. The simulations were conducted one-by-one for each regolith type, in the same manner as described earlier for particle LoGSP-1, but this time for three different launch sites.

7.4.1 REGOLITH LOFT FROM LONGEST EDGE OF ASTEROID

Figure 7.33 shows the number of particles that resulted in re-impact after launch, for each particle type and initial Solar phase angle, against the launch velocity. Information about launch azimuth can not be derived from this plot. The overall trend³ remains the same across all plots for different initial Solar phases. There are small changes in the absolute numbers but they are significantly small and do not affect the qualitative property. From Figure 7.33 we can infer that presence of Solar perturbations does not drastically affect the re-impact behavior of different regolith types.

Figure 7.34 shows the x-y plane trajectories, an example that results in a re-impact scenario, for all regolith types mentioned in Table 7.1. The particles have identical launch conditions for each initial Solar phase angle. What we witness from the plots is that for the same launch conditions that lead to a re-impact scenario, the trajectories remain almost similar and are not affected much by changes in regolith size and density. The similarity is also observed across different initial Solar phase angles. This implies that the perturbations are not significantly affecting the trajectories. We are looking at an isolated example here, but the fact that perturbations are not drastically affecting the re-impact trajectories is true to a very large extent and can be witnessed in the re-impact maps.

Two re-impact maps are shown in Figure 7.36 and Figure 7.35. The former is for regoliths with the same density of 3.2 g/cm^3 but particle radii 1 cm and 5 cm; and the latter is for regoliths of the same particle radius of 1 cm and densities 3.2 g/cm^3 and 7.5 g/cm^3 . What we see first hand is that in majority of the cases, the re-impact locations for different regolith types are overlapped. For the fewer cases where different particles have different re-impact locations, we see that the corresponding locations are very close to each other. This is also seen from the ending points on the trajectory plots in Figure 7.34a and Figure 7.34d. For even more rarer cases (wherein a common launch condition did not result in all particle types re-impacting), it was observed that the particles re-impacted at locations far away from each other. But from the majority, we can generalize that re-impact locations do not change significantly between different regolith types (used in this simulation and thesis) and/or with different initial Solar phase angles. Similar re-impact maps for regoliths with the density of 7.5 g/cm^3 but particle radii 1 cm and 5 cm; and for regoliths of the same particle radius of 5 cm and densities 3.2 g/cm^3 and 7.5 g/cm^3 , are shown in Figure C.9 and Figure C.8 respectively.

³The trend over here is judged by looking at which grid cell the bars in the plot end up in.

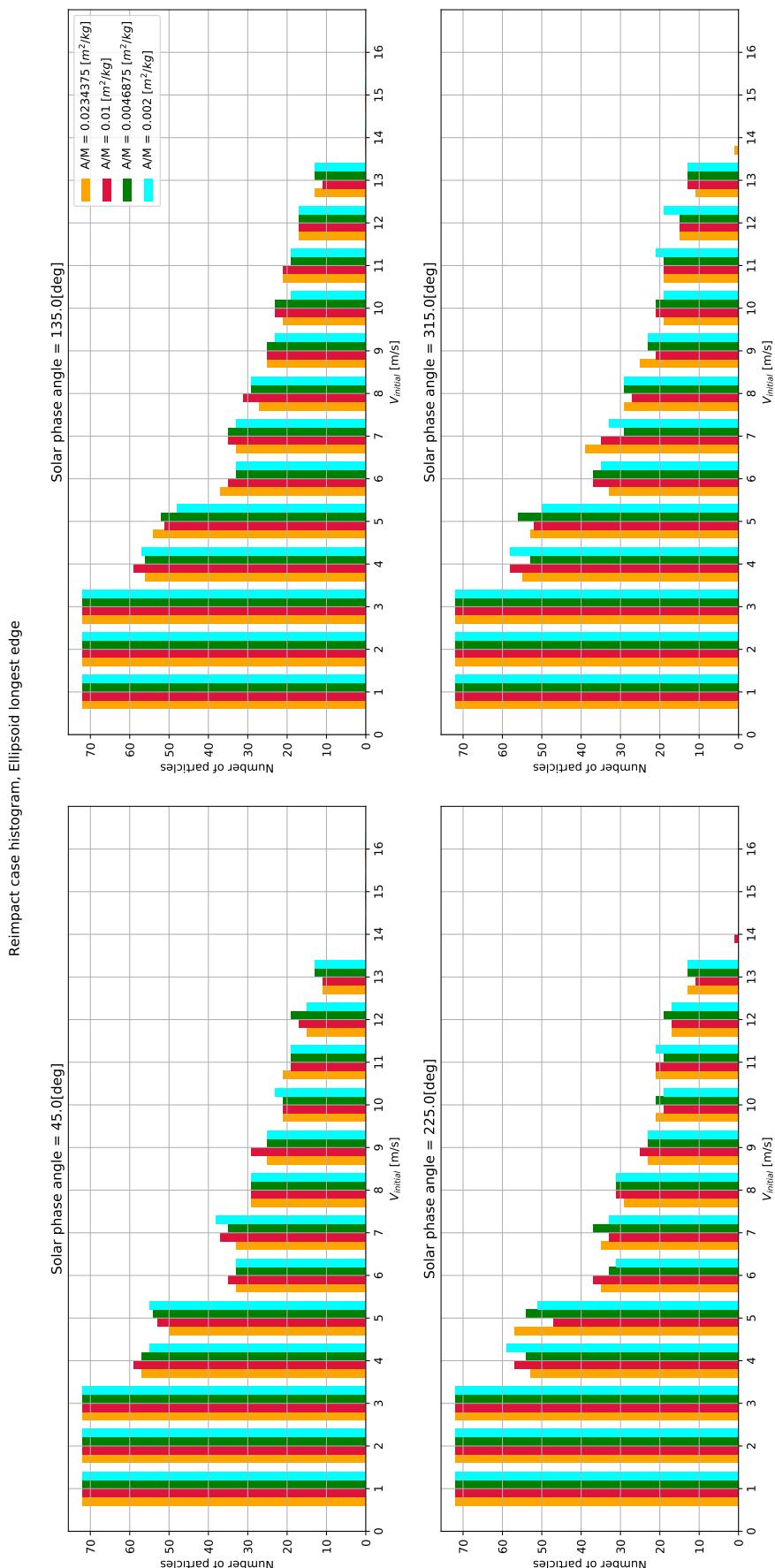


Figure 7.33: Number of re-impact cases for all regolith types launched from the longest edge of the asteroid. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

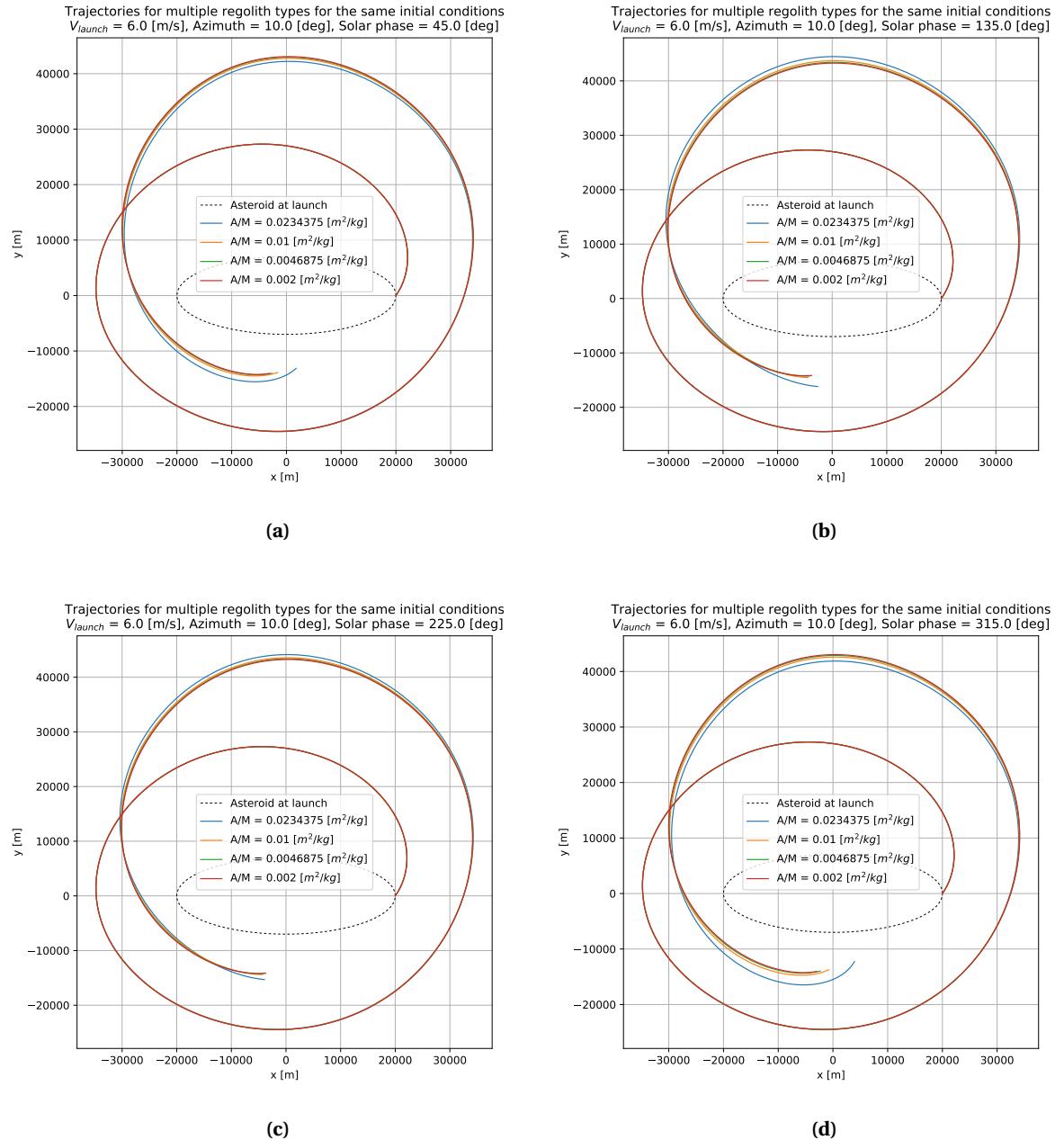


Figure 7.34: Re-impact trajectories for all regolith types mentioned in Table 7.1. The trajectories are shown for different initial Solar phase values but for each, the launch conditions are identical for all particle types. The trajectories are expressed in the AIF and the point where the trajectories abruptly end is where the re-impact occurs with the rotating asteroid.

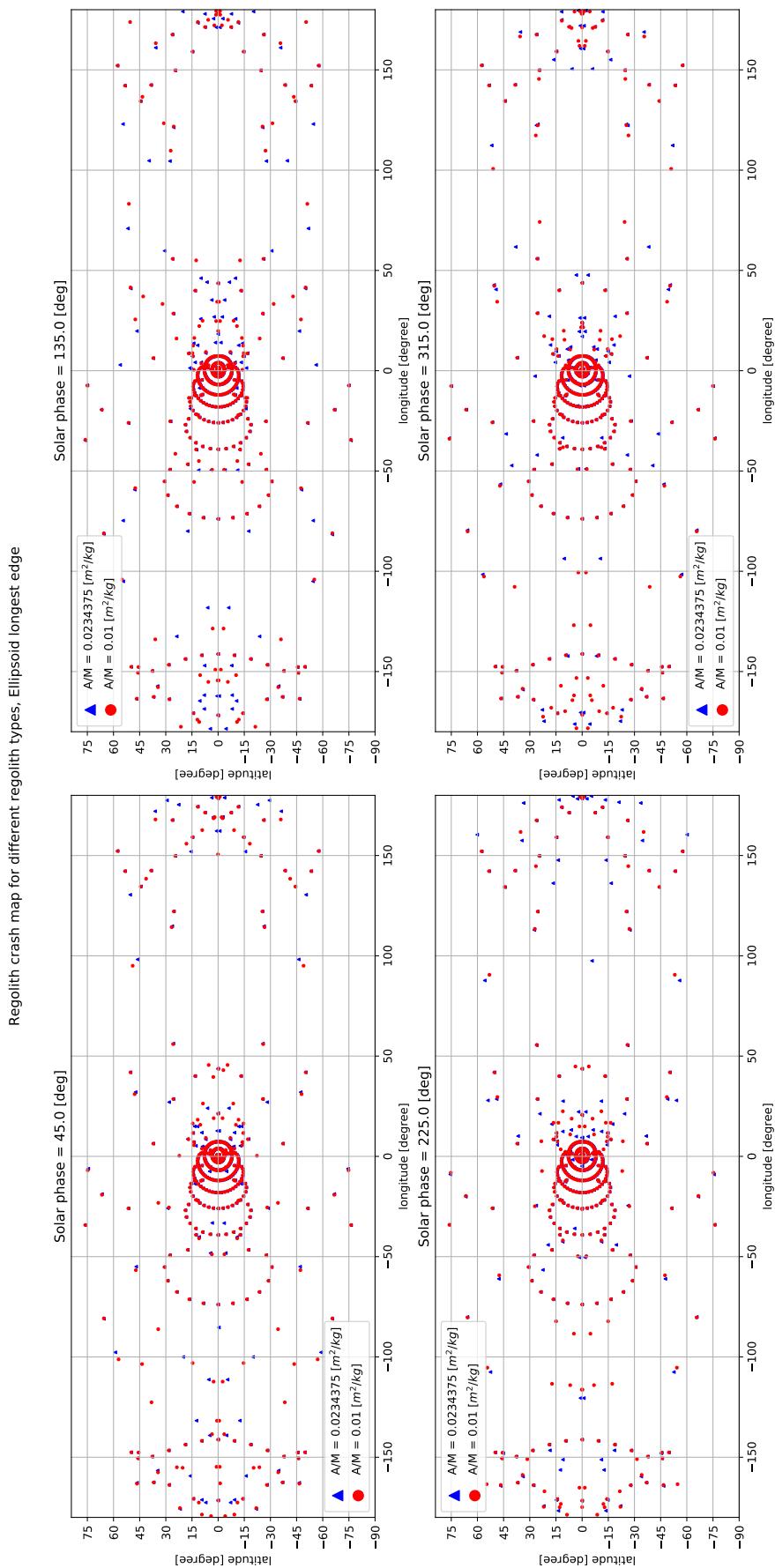


Figure 7.35: Re-impact locations for regoliths of different density but same size (i.e 1 cm). The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

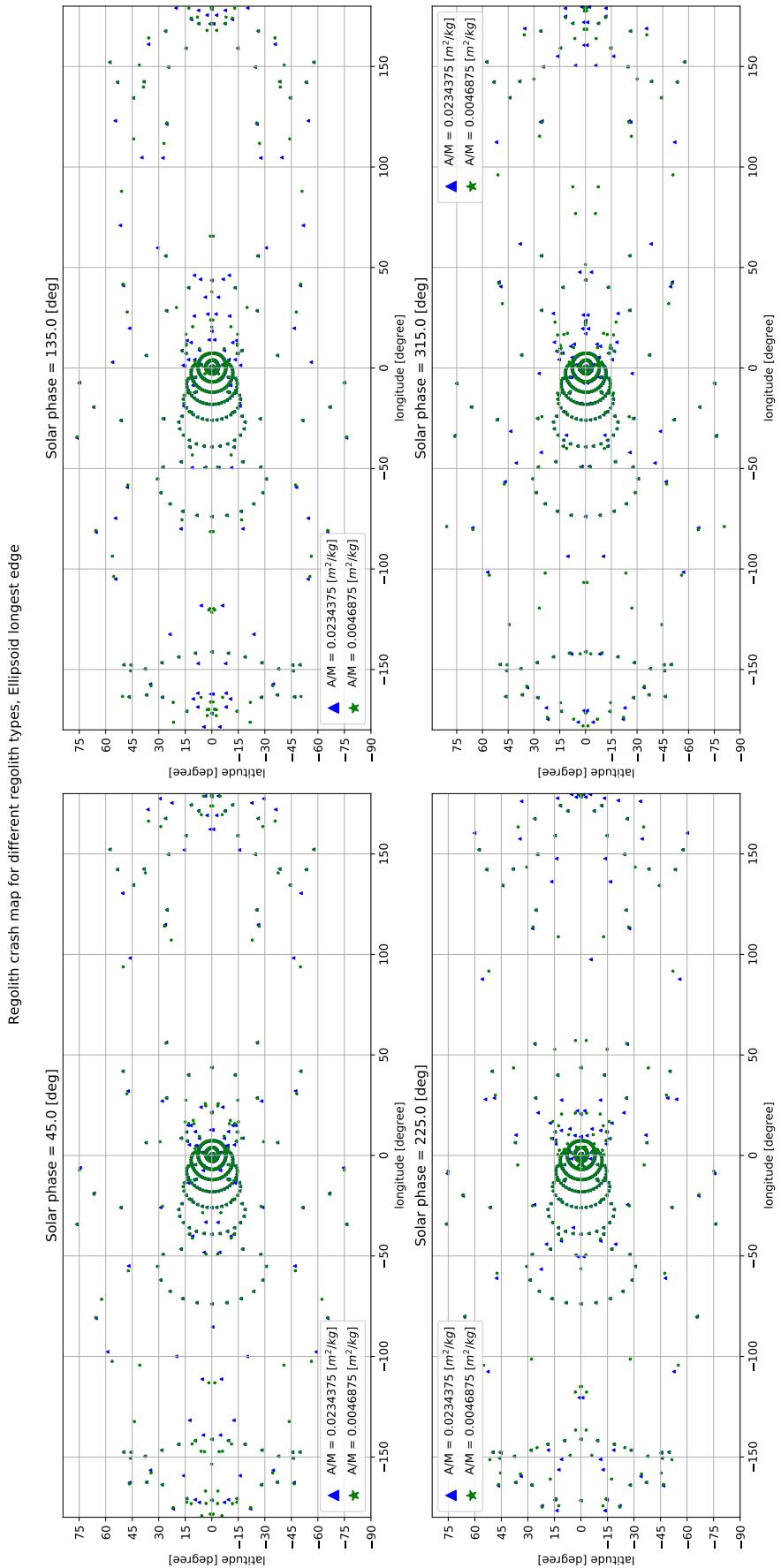


Figure 7.36: Re-impact locations for regoliths of different sizes but the same density (i.e. 3.2 g/cm^3). The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

Figure 7.37 shows the number of cases that lead to an escape situation for all the regolith types and for different initial Solar phase angles. Just like that for the re-impact case, what we observe here is that the general trend of the number of cases does not vary much with different initial Solar phase angles. Although, the fluctuations in exact numbers are clearly visible. For extremely high velocities, 15 m/s and beyond, the particles always escape and are not affected by the presence of the Solar perturbations.

Figure 7.38 shows the variation in the HEV with launch azimuth, for three different launch velocities and for an initial Solar phase angle of 45°. For the remaining Solar phases, the plots are similar and can be found in the appendix in Figures C.10 to C.12. In the plot for velocity 13 m/s we see that the escape behavior is identical for all particle types as the data points for each overlap one another. As previously discussed, the relationship between the distribution of data points in a smooth and continuous form and the escape behavior and trajectory, is that the particles are on an escape trajectory immediately after launch and hence do not require an assist from the rapid rotation of the asteroid. The latter is required, for example, in the case of 5 m/s as observed from the uneven distribution of the points. Thus for smaller velocities, the HEV data points for different regolith types are not overlapping, suggesting that the perturbations have a significant effect on their trajectories. But as the velocities increase, the perturbations have lesser of an effect on the trajectories. Another way of saying this is that when the launch velocity is itself large, the particles are on a relatively quicker escape trajectory, which means that the perturbing acceleration does not surmount to a value that can cause significant changes. This is also observed in the trajectory plots for all the particles.

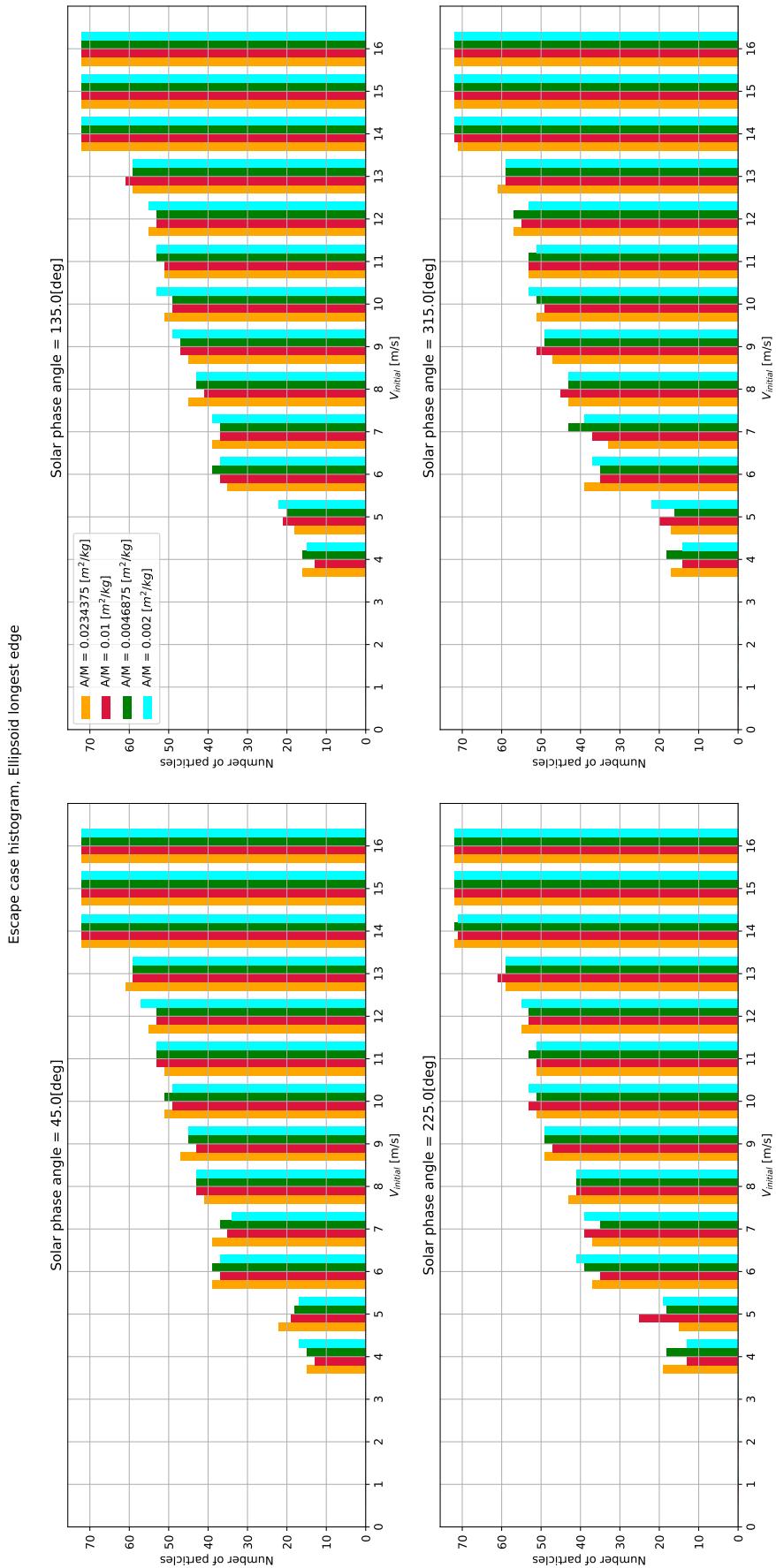


Figure 7.37: Number of all escape cases for all regolith types launched from the longest edge of the asteroid. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

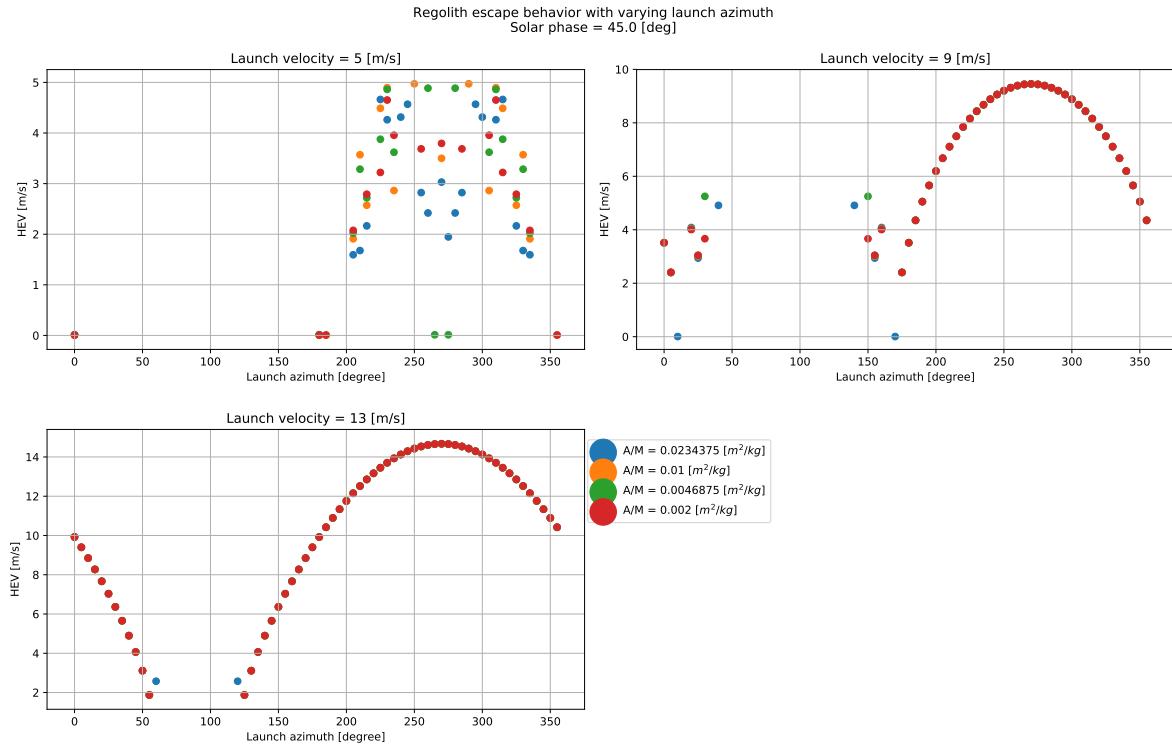


Figure 7.38: HEV, shown for all regolith types used in this thesis, but only for three specific launch velocities. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

Figure 7.39a shows the trajectories for a launch velocity of 5 m/s, launch azimuth of 225° and an initial Solar phase angle of 45°. Figure 7.39b shows the trajectory for a launch velocity of 9 m/s with all other initial conditions being the same. We see that for a higher velocity, all regolith types have the same escape behavior or trajectory and are hardly affected by the Solar perturbations. For the smaller velocity, we see that all particles are distinctly separated as they embark on the escape trajectory in the end. The difference is significantly large. A zoomed in version of the trajectory plot for the case of 5 m/s is shown in Figure C.13. For the example case depicted in Figure 7.39a, and keeping in mind that the initial Solar phase angle is 45°, the differences in the trajectories start emerging slowly much earlier in the simulation (before even the first revolution is completed), with the highest area-to-mass ratio particle getting pushed outwards (due to a higher SRP effect) relative to the one with the lowest area-to-mass ratio. As explained previously for the capture scenario, in this case too, we witness that different regolith types are subject to different phasing with respect to the asteroid. Thus as the particles approach the vicinity of the asteroid, the existing trajectory differences (or phase differences) get amplified by the rapid rotation of the asteroid, leading to 4 distinct escape routes.

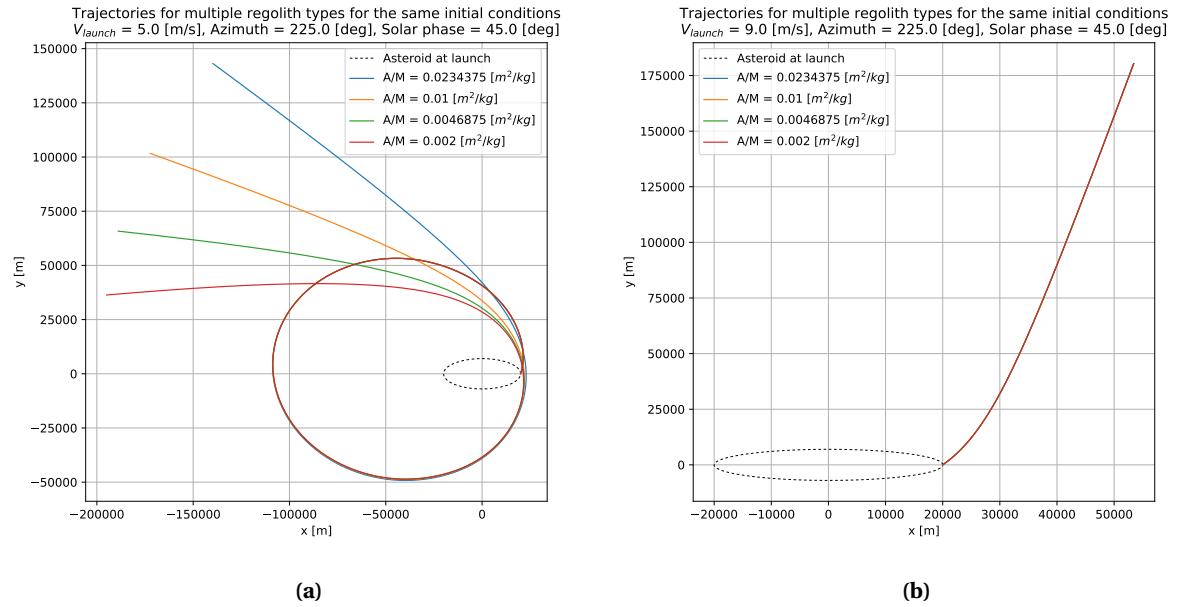


Figure 7.39: Escape trajectories for all regolith types, highlighting the diminishing effect of Solar perturbations as the launch velocity increases (keeping all other initial conditions same). The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

Figure 7.40 shows the number of capture cases for all regolith types. The number of capture cases are extremely small relative to the number of escape and re-impact cases shown previously. We will look at the trajectory of an example capture case from Figure 7.40, i.e. particle LoGSP-4, at launch velocity 6 m/s, launch azimuth of 20°, and initial Solar phase angle of 45°. For the same launch conditions, the trajectories for the remaining regolith types were also plotted, neither of which resulted in capture, and are shown in Figure 7.42. A zoomed in version of the trajectory, highlighting the visible separation in the regolith trajectories is shown in Figure 7.41. The reasons for separation of the trajectories for the different regolith types, in this situation, is the same as explained recently for the escape scenario. Due to Solar perturbations, the trajectory for each particle is shifted such that their phase with respect to the rotating asteroid. The only difference here, is that all particles don't share the same fate with each other.

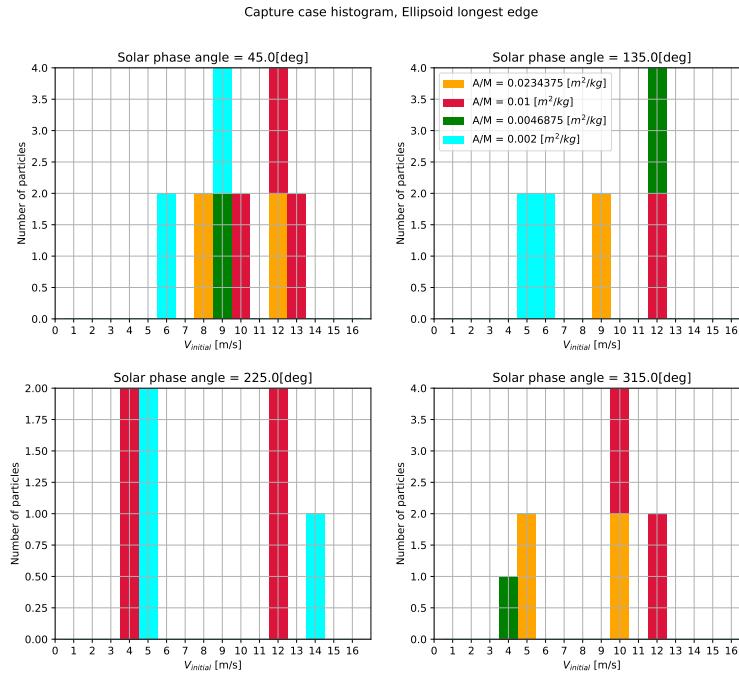


Figure 7.40: Number of all capture cases for all regolith types launched from the longest edge of the asteroid. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

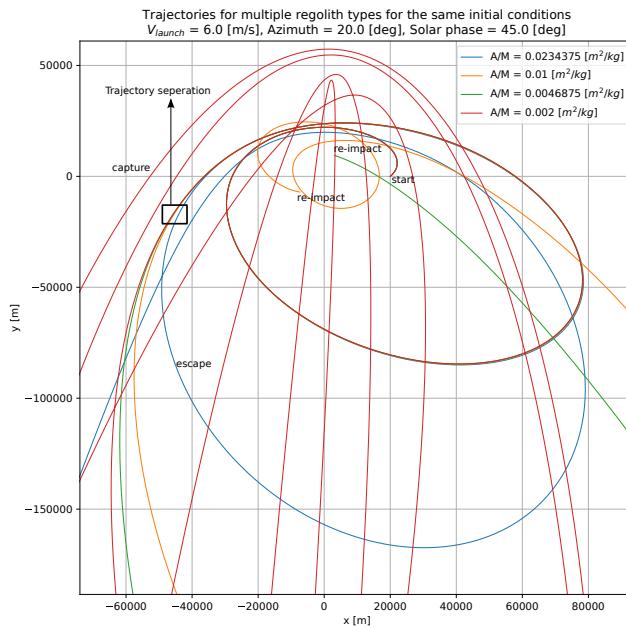


Figure 7.41: Zoomed-in version of the plot shown in Figure 7.42. This view enhancement highlights the instance when the trajectories begin to visibly separate from each other. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

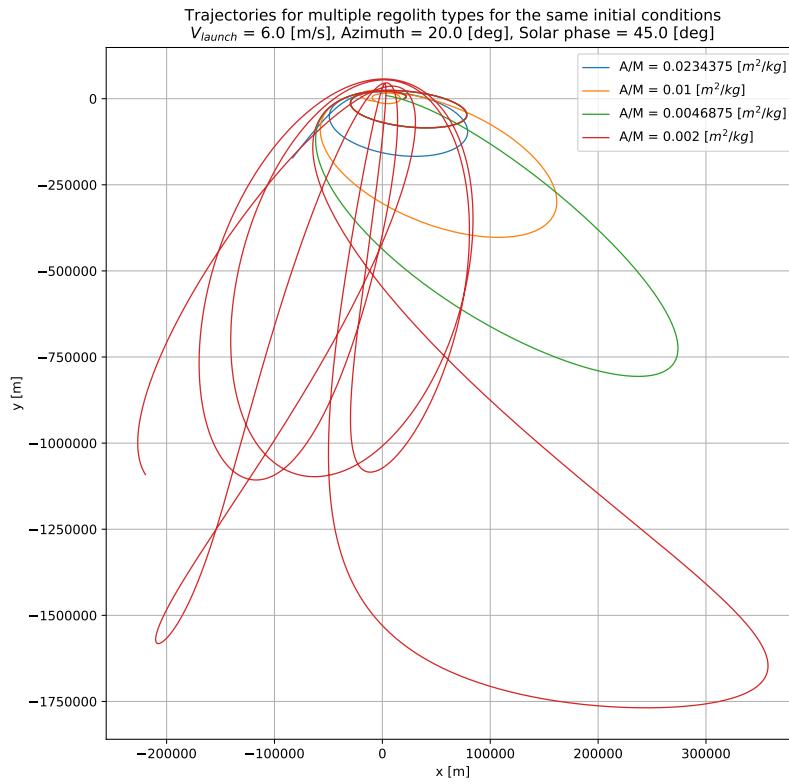


Figure 7.42: Trajectories plotted for all regolith types, with launch conditions that lead to a capture case only for particle LoGSP-4. For particle LoGSP-1, the final fate is escape and for LoGSP-2 and LoGSP-3, the final fate is re-impact. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

7.4.2 REGOLITH LOFT FROM LEADING EDGE OF ASTEROID

Figure 7.43 shows the total number of re-impact cases for all regolith types launched from a fixed location on the leading edge of the asteroid. The number of re-impact cases are far more than those observed for the longest edge scenario. Even for a relatively high velocity as 11 m/s, all particles result in a re-impact. Their are two reasons for this behavior. The first is that the gravitational attraction for the current launch location is stronger than that on the longest edge of the asteroid. The latter has the weakest gravitational attraction on the entire ellipsoid shaped asteroid. The second is that the particles have a much lower inertial velocity at rest (due to the asteroid's rotation) at the current launch site as it is radially closer (from the centre of the asteroid) to the axis of rotation, compared to that for a launch site at the longest edge of the asteroid. Thus, we see an increase in the number of re-impact cases and especially for the higher velocities.

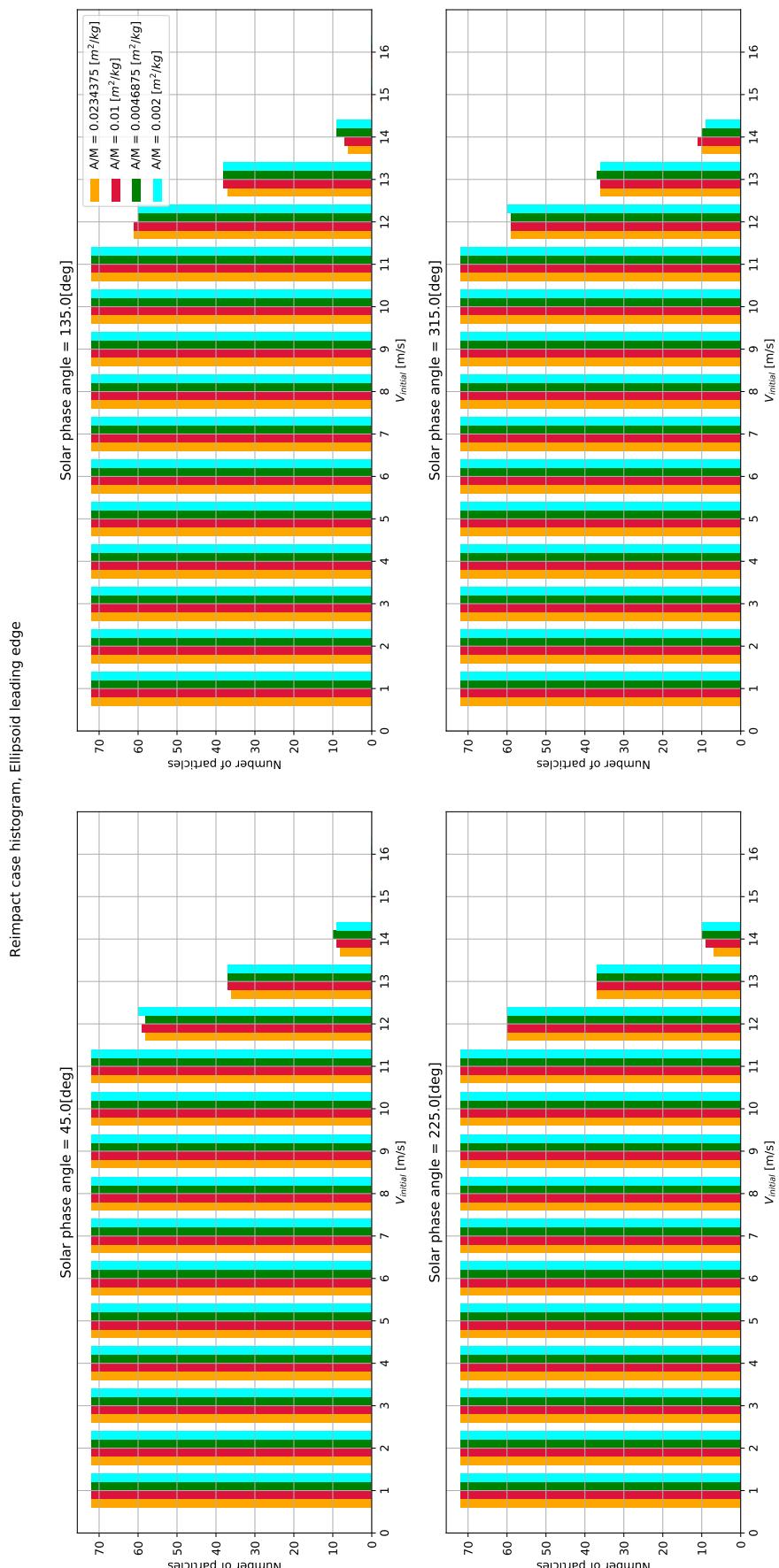


Figure 7.43: Number of all re-impact cases for all regolith types launched from the leading edge of the asteroid. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

The presence of Solar radiation pressure has no significant effect on the particle trajectories. Compared to the longest edge, there are fewer cases where different regolith types have distinct re-impact locations. We observe this in the re-impact maps shown in Figure 7.44, where regoliths have the same particle radii of 1 cm and densities 3.2 g/cm³ and 7.5 g/cm³; and Figure 7.45, where regoliths have the same density of 3.2 g/cm³ and particle radii of 1 cm and 5 cm. The number of distinct re-impact locations between different regolith types is certainly less than that for particles launched from the longest edge. However, compared to the latter, the re-impact locations are more widespread. From the re-impact maps, we also observe that the (apart from a very few individual cases) the re-impact locations remain the same for different initial Solar phase angles.

Figure 7.46 shows the re-impact trajectories for all four regolith types, launched at velocity 11 m/s, azimuth 355° and initial Solar phase angle 45°. The trajectories for all particles is exactly the same which means that the perturbations have absolutely no effect on them at all. Figure 7.47 shows another example of re-impact trajectories for all the regoliths, launched with the same initial conditions, i.e. a velocity of 14 m/s, azimuth 155° and initial Solar phase angle 45°. In this particular case, we can see that the particle with the highest area-to-mass ratio is being affected the first as well as the most by the perturbations. The direction in which the particle trajectories start getting separated is also consistent with the initial apparent position of the Sun. Also note that in this case, keeping all other initial conditions the same, a re-impact scenario is not always obtained for all regolith types for different initial Solar phase angle. For example in the case of initial Solar phase being 135°, the trajectory plot for which is shown in Figure 7.48. Note that, distinct trajectories for each regolith type are still observed, and in that, particles LoGSP-1 and LoGSP-2 are the ones who have escape as their final fate while the remaining two have re-impact. For the launch velocity of 14 m/s and azimuth 115°, and for the remaining two Solar phase angles, all particle types re-impact and each have distinct trajectories facilitated by the Solar perturbations. They can be found in Figures C.14 and C.15.

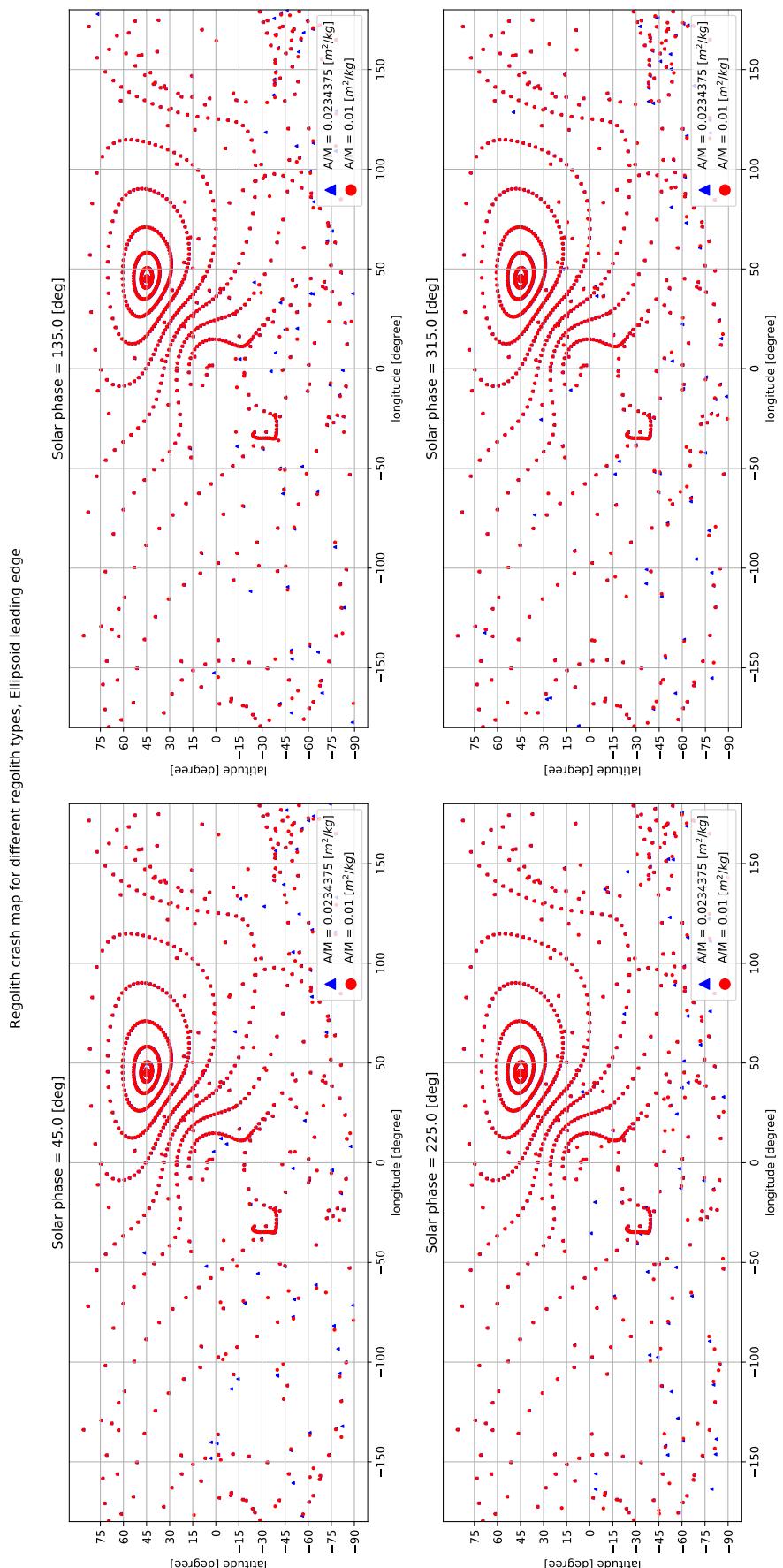


Figure 7.44: Re-impact locations for regoliths of different density but same size (i.e 1 cm). The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

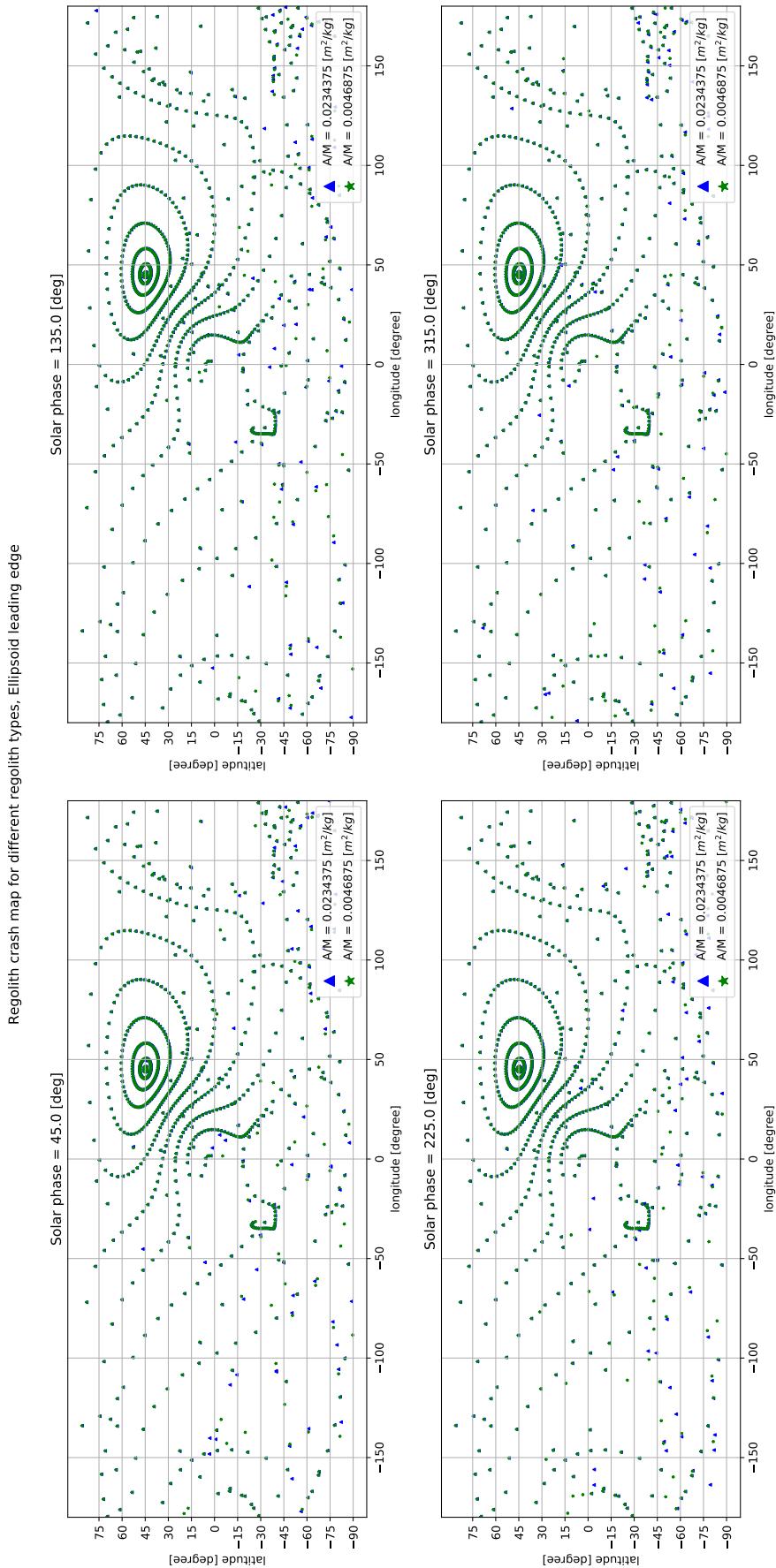


Figure 7.45: Re-impact locations for regoliths of different sizes but the same density (i.e. 3.2 g/cm^3). The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

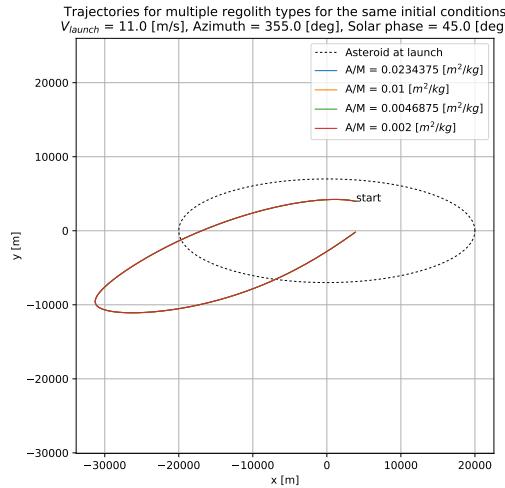


Figure 7.46: Re-impact trajectory for all regolith types used in this simulation, each launched with the same initial conditions. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

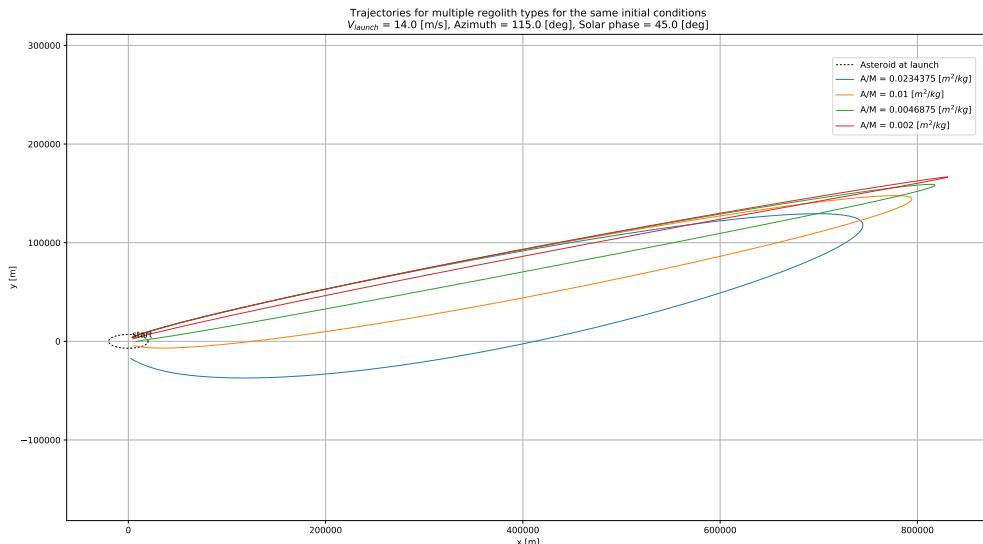


Figure 7.47: Re-impact trajectory for all regolith types used in this simulation, each launched with the same initial conditions. In this case the particle trajectories are visibly separated from each other because of the Solar perturbations.

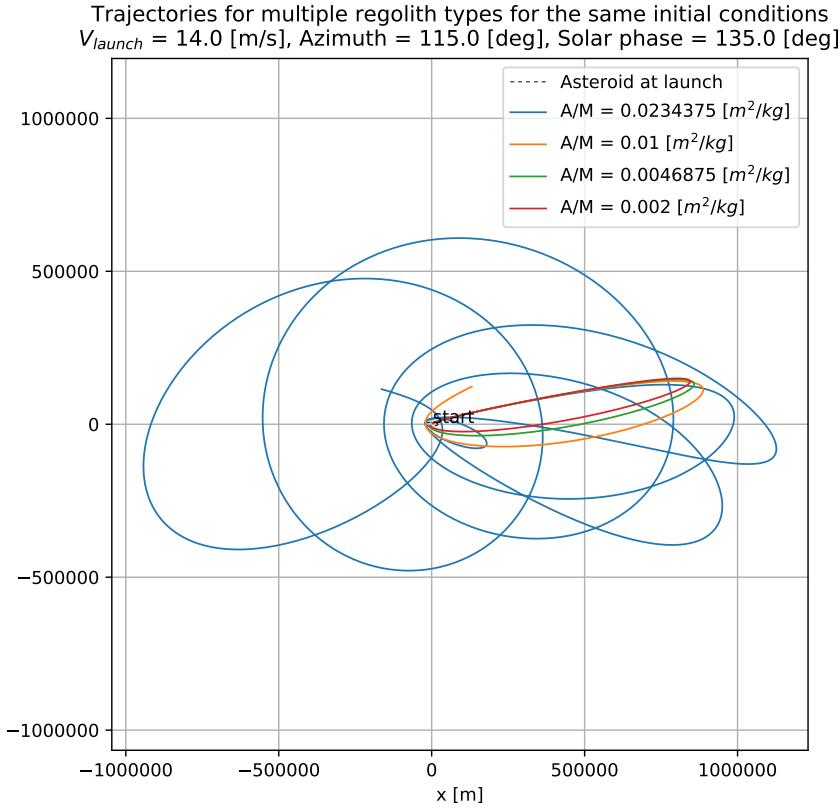


Figure 7.48: Both escape and re-impact behavior is observed in this simulation, although, each regolith type was launched with the same initial conditions. However, particle trajectories are still visibly separated from each other because of the Solar perturbations.

Figure 7.49 shows the number of escape cases which are extremely small relative to the case of particles launched from the longest edge. Figure 7.50 shows an example case of HEV for all regolith types, but for only three distinct launch velocities and an initial Solar phase angle of 225°. The behavior was found to be similar for all other initial Solar phase angles as well and hence is not shown here for brevity. Just like for the case of longest edge, we test the relationship between the manner of distribution of data points and the escape behavior and whether it is found to be true here as well. We examine the eccentricity and energy plots and the trajectory plots for all regoliths of type LoGSP-4 that escaped and had an initial velocity of 12 m/s in Figure 7.50. The results are unlike that in the case of the longest edge. The eccentricity and energy (shown in Figure 7.51) both indicate that the particle are not on an escape trajectory immediately after launch despite the even and continuous distribution of HEV data points. However, we do notice a pattern in the trajectory plots shown in Figure 7.52. For the given even distribution of HEV data points, the escape trajectories do not involve multiple revolutions around the asteroid as long as the HEV is above 0 m/s. This behavior of escape trajectory revolution and corresponding energy and eccentricity was also observed for particle LoGSP-1 and for the same launch velocity case. The plot for its trajectories is shown in Figure C.17 and for the energy and eccentricity is shown in Figure C.16.

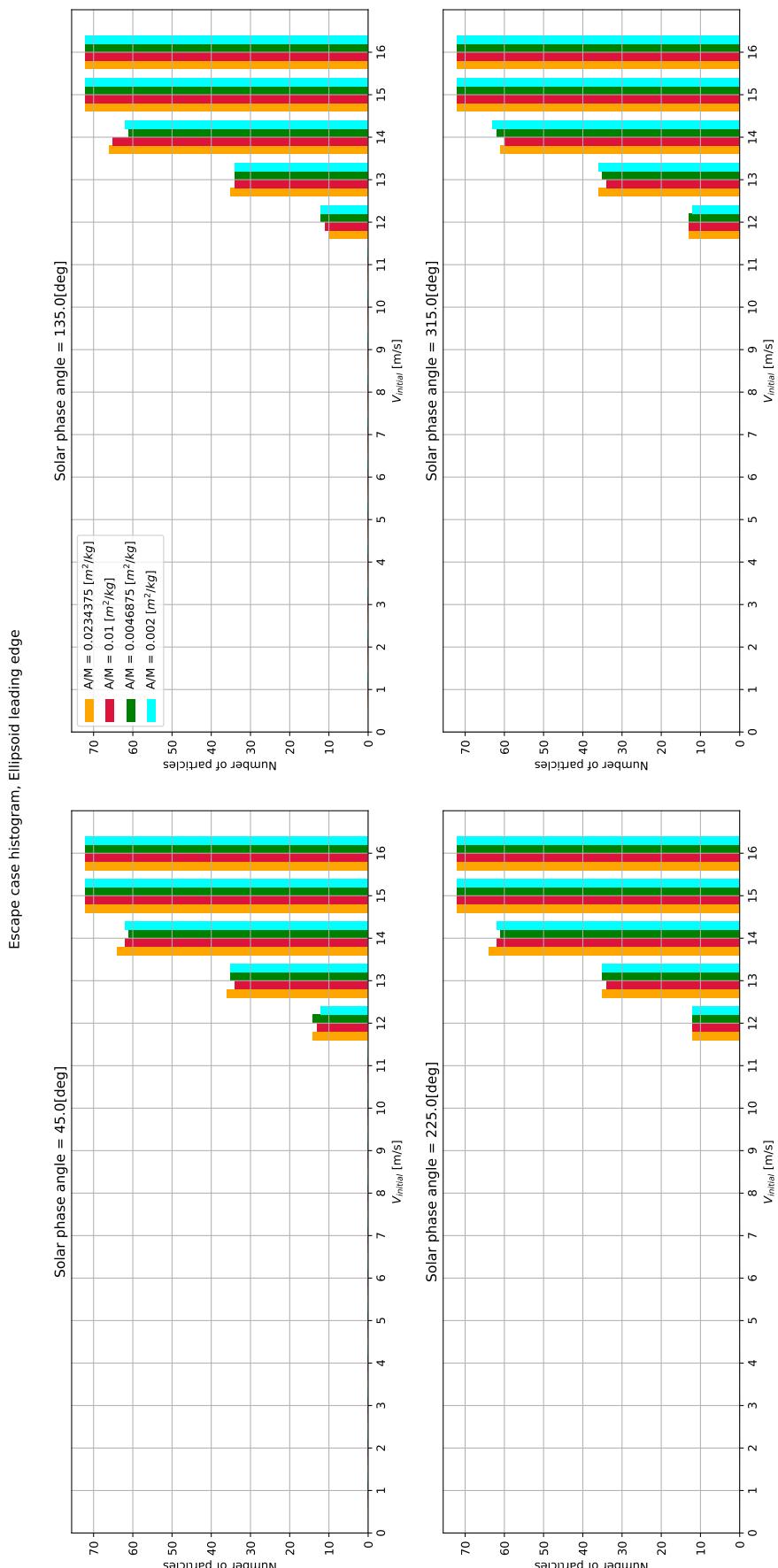


Figure 7.49: Number of all escape cases for all regolith types launched from the leading edge of the asteroid. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

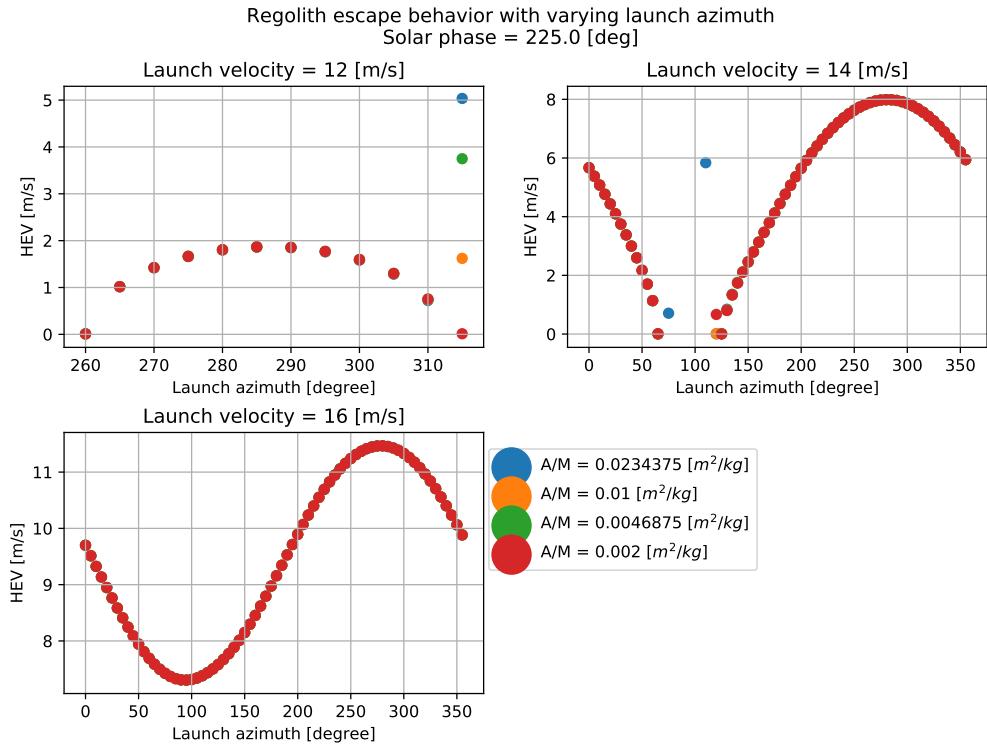


Figure 7.50: Plot depicting HEV for three distinct launch velocities, and for all regolith types launched from the leading edge of the asteroid, for an initial Solar phase angle of 225°. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

Now let's consider the scenario again for particle LoGSP-1 for a launch velocity of 14 m/s and launch azimuths 30° - 100° in Figure 7.50. The energy and eccentricity plots for these are given in Figure C.18 and the trajectory plots are shown in Figure C.19. Again the behavior in these plots is the same as that explained earlier for the case of 12 m/s. But, for the same particle and launch velocity, if we now consider launch azimuths 200° to 250°, we notice that the particle trajectories are hyperbolic immediately after launch (see Figure C.21 for trajectories and Figure C.20 for energy and eccentricity evolution), which is unlike what we have seen so far for the leading edge. Thus, the only thing we can perhaps generalize in the case of a launch site on the leading edge, with regards to the escape behavior and relation to the distribution of the HEV data points, is the number of orbital revolutions attained by the particle around the asteroid before it escaped.

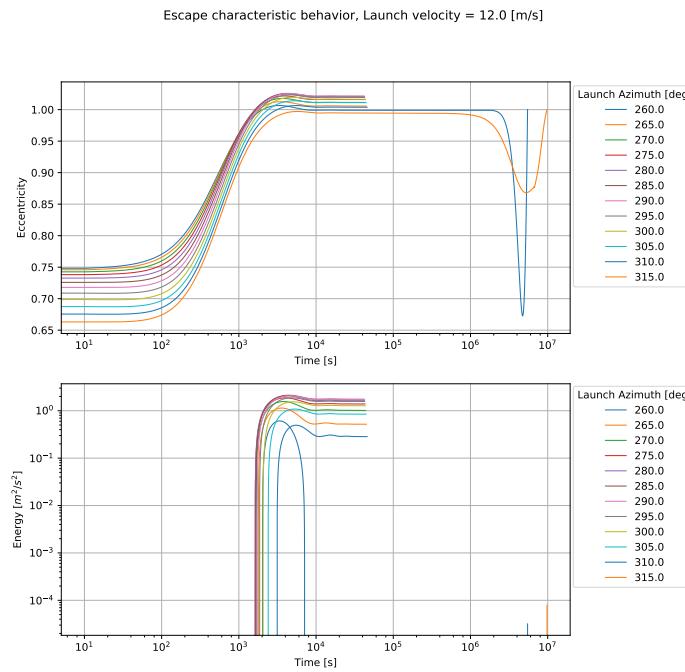


Figure 7.51: Plot depicting the variation in energy and eccentricity with time for all escape cases for particle LoGSP-4 launched with a velocity of 12 m/s from Figure 7.50.

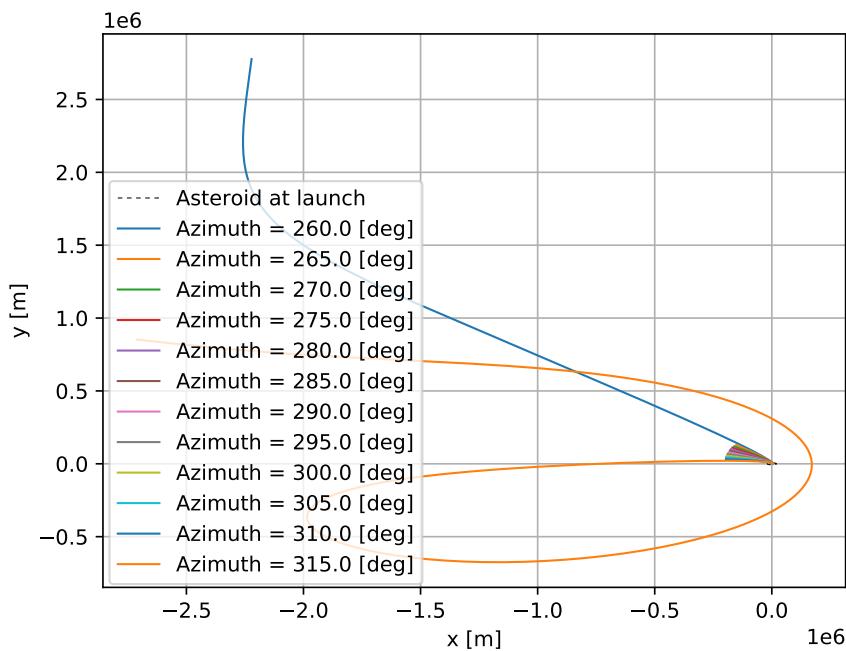


Figure 7.52: Plot depicting the trajectory evolution for all escape cases for particle LoGSP-4 launched with a velocity of 12 m/s from Figure 7.50.

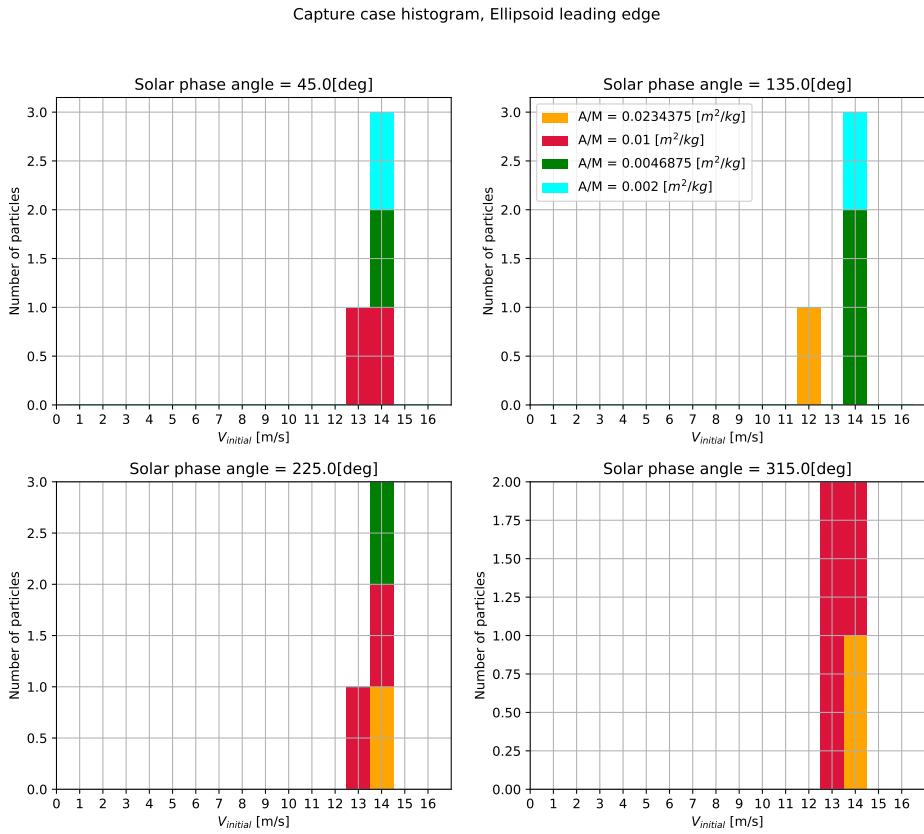


Figure 7.53: Number of all capture cases for all regolith types launched from the leading edge of the asteroid. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

Finally, we arrive at the capture cases for particles launched from the leading edge launch site. The number of cases are depicted in Figure 7.53. The number of capture cases are lesser relative to those obtained in the longest edge scenario, although, atleast one capture case is obtained for each regolith type and a majority of them are obtained at the launch velocity of 14 m/s. Looking at an example case, for initial Solar phase angle of 225°, for azimuth angle of 70°, capture orbits are obtained for particles LoGSP-1 and LoGSP-2. For the same initial conditions, particles LoGSP-3 and LoGSP-4 result in re-impact. The trajectory for all four particles, for the given initial conditions is shown in Figure 7.54. Note, that the box highlighting the trajectory separation for all four regolith types, and the subsequent motion of LoGSP-1 and LoGSP-2 is consistent with the direction of the Solar perturbations (given that the initial Solar phase is 225°). The perturbations are not, however, as effective in drastically changing the particle trajectories for LoGSP-3 and LoGSP-4, which is clearly visible in the plot.

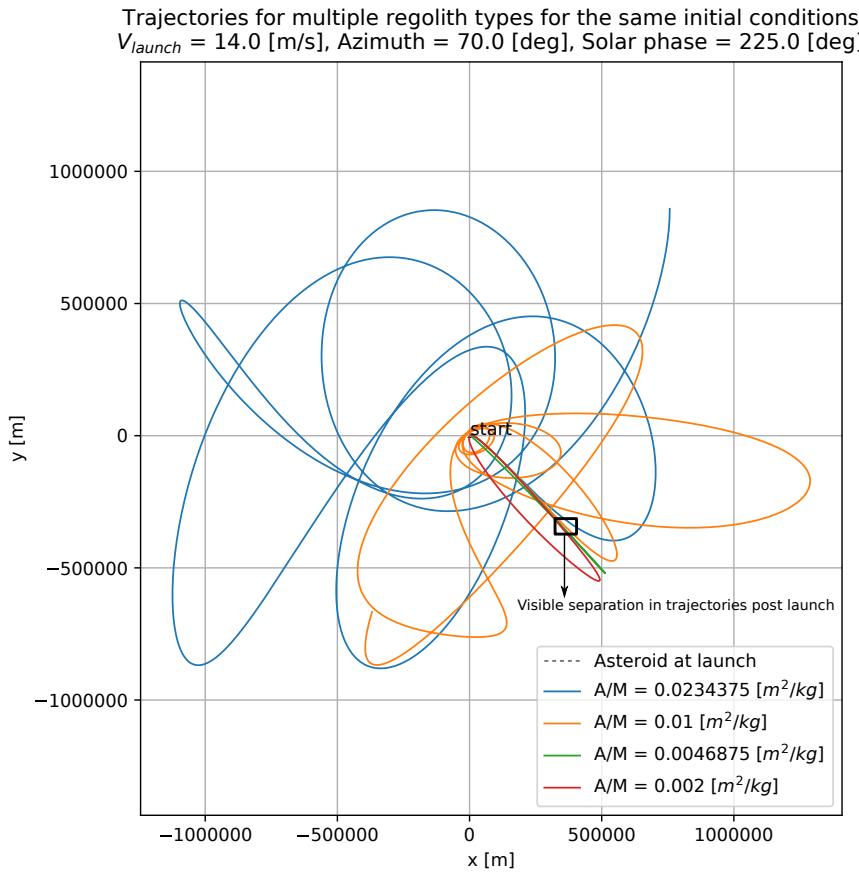


Figure 7.54: For the given initial conditions, this plot shows capture trajectories for particles LoGSP-1 and LoGSP-2. For the remaining two particle types, we witness a re-impact scenario.. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

7.4.3 REGOLITH LOFT FROM TRAILING EDGE OF ASTEROID

Figure 7.55 shows the histogram plot for the re-impact scenario for particles launched from a fixed launch site on the trailing edge of the asteroid. For the trailing edge, we have even more number of re-impact cases than those for the leading edge scenario. A few re-impact cases are observed even for a high launch velocity of 15 m/s. And the reason for this is obviously the location of the launch site. If we go back to Figure 4.8, one would notice that the surface inertial velocity vector points into the surface of the asteroid, instead of going out like that for the leading edge. Thus for trailing edge launch sites, the particle net inertial velocity is much more reduced than that for the leading edge, which is why we see more re-impact cases for the former, and especially at higher launch velocities. Just like with the launch sites at the longest and the leading edges, the general re-impact behavior here remains the same across different initial Solar phase angles.

A peculiar behavior is noticed for launch velocity 9 m/s where all particles don't have re-impact as their final fate, few of which have either escape or temporary capture as their final fate. We take a look at an example for particle LoGSP-1, where it is launched at an azimuth of 270°, initial Solar phase angle of 45° and launch velocities of 8, 9 and 10 m/s. The trajectory plot is given in Figure 7.56a. For the same particle launched at 8 and 10 m/s, the final fate is re-impact (for the given azimuth and Solar phase angle). However, at 9 m/s we see that the particle escapes. From Figure 7.56b, which shows the trajectories for all regolith types launched with the same initial conditions as before but at 9 m/s, we know that Solar perturbations are not responsible for the aforementioned behav-

ior because if they were then we would have seen noticeable differences between the trajectories for each regolith type. Thus, in Figure 7.56a, at 9 m/s, the particle achieves a phase relative to the asteroid such that it acquires an assist from the asteroid's rotation and escapes, thereby avoiding a re-impact scenario. This type of behavior, however, wouldn't in general be restricted only to one specific velocity even though that's what we witness here and the reason for that is the resolution of launch azimuth being varied coarsely at 5°. Its possible that for a finer resolution, a different trend be observed in the histogram plots.

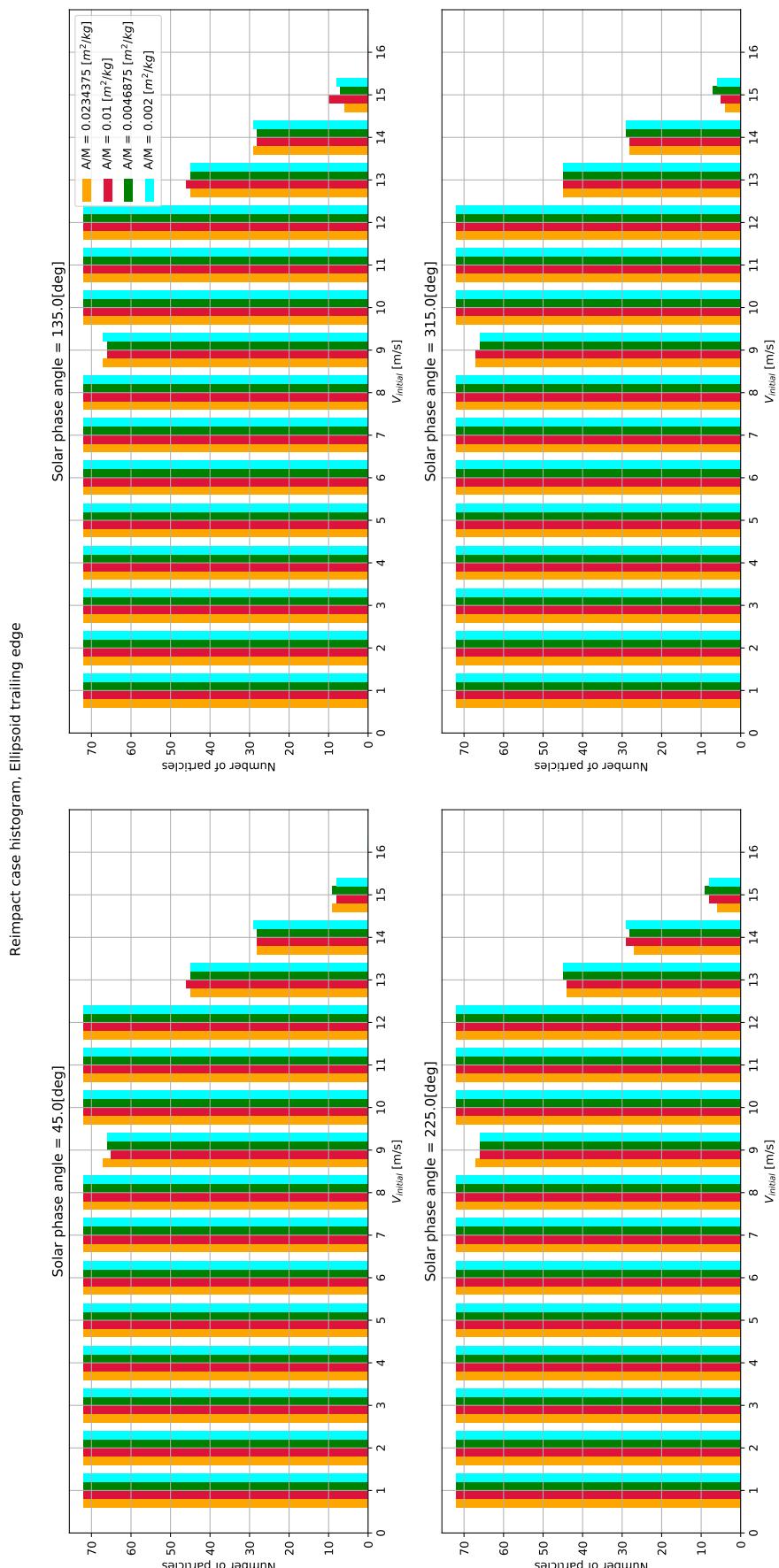
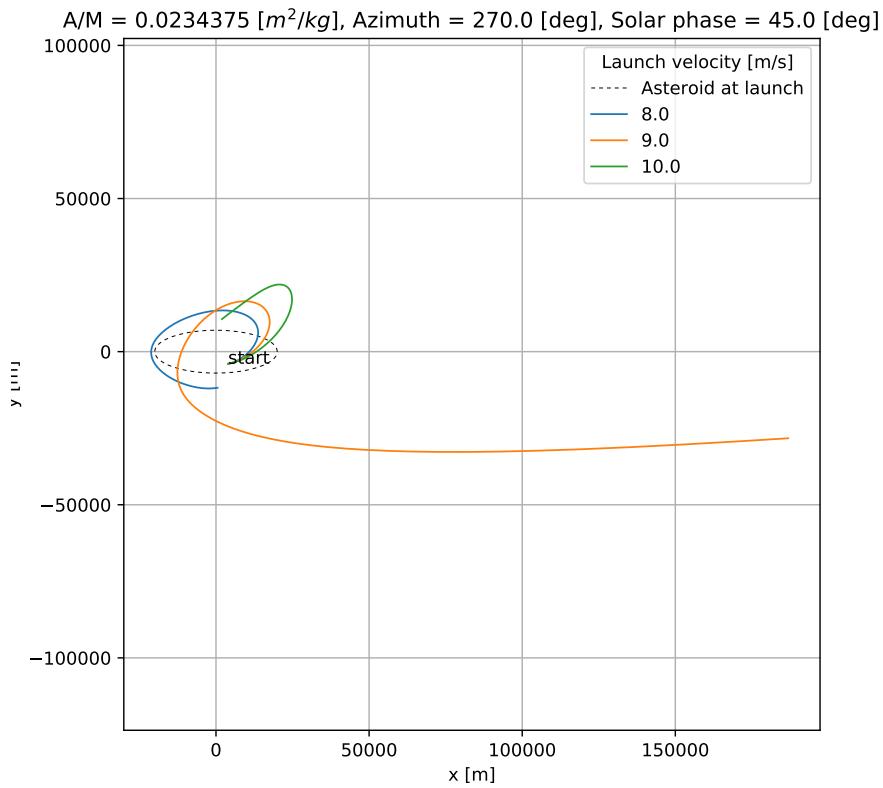
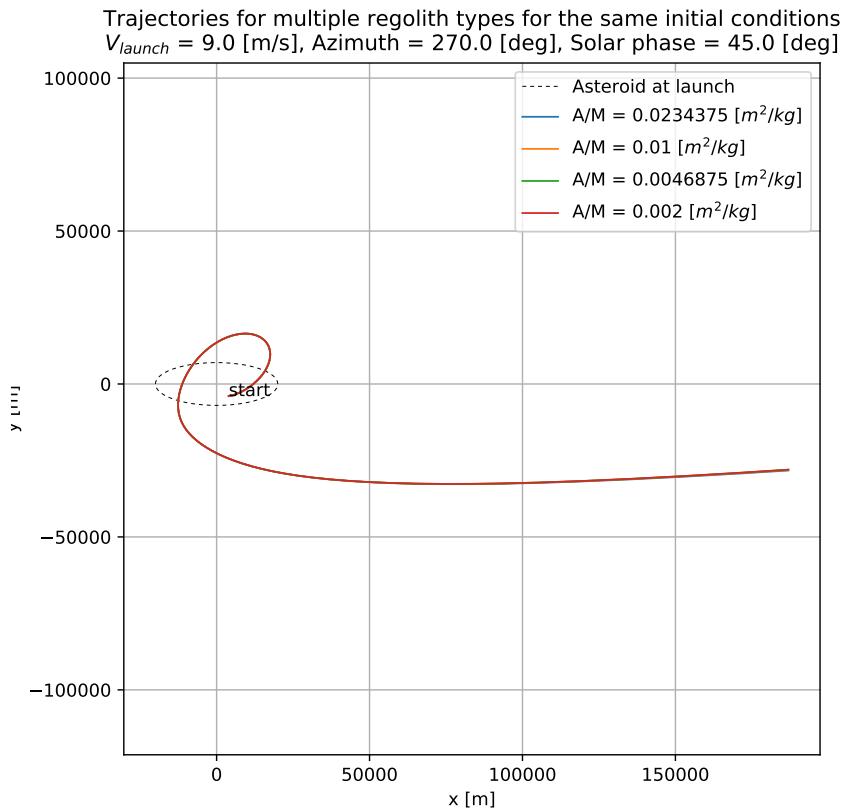


Figure 7.55: Number of all re-impact cases for all regolith types launched from the trailing edge of the asteroid. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.



(a)



(b)

Figure 7.56: (a) Trajectories for particle LoGSP-1 at three different velocities, but the same launch azimuth and initial Solar phase angle. (b) Escape trajectories for all regolith types for the launch velocity of 9 m/s.

Figure 7.57 shows the re-impact locations, for regoliths with density 3.2 g/cm^3 and 7.5 g/cm^3 and particle radius of 1 cm, once lofted from the launch site at the trailing edge of the asteroid. Since the number of re-impact particles is higher than its leading edge counterpart, especially for the higher launch velocities, we see a slightly larger number of distinguishable particles based on their re-impact locations. Apart from a few isolated cases in the mid-range launch velocities, it was observed that only at higher launch velocities, when the particles are launched into time consuming orbits, that the Solar perturbations become effective in separating the trajectories which ultimately led to the few distinguished re-impact locations that we observe in the map. Again, just like previous launch scenarios, we also observe that the re-impact maps in this case do not vary drastically with different initial Solar phase angles.

A similar situation is obtained for the case when the density for regoliths is kept the same at 3.2 g/cm^3 and particle radii at 1 cm and 5 cm. The re-impact map for the case is shown in Figure 7.58. Figure 7.59 shows an example of re-impact trajectories for all regolith types for the same initial conditions. Note the separation of the trajectories. The particle with the highest area-to-mass ratio is again affected the most relative to others. All particles have overlapping trajectories soon after launch but eventually separate into four distinguishable trajectories that re-impact at wildly different locations. The direction of separation is also consistent with the initial Solar phase angle, where the particle with the highest area-to-mass ratio getting pushed outwards the most. Similar behavior was noted across other initial Solar phase angles as well, keeping all other initial conditions the same. The trajectory plots for them can be found in Figures C.22 to C.24. In general this behavior of re-impact trajectories for the trailing edge case was observed for other initial conditions as well but only for the higher launch velocities.

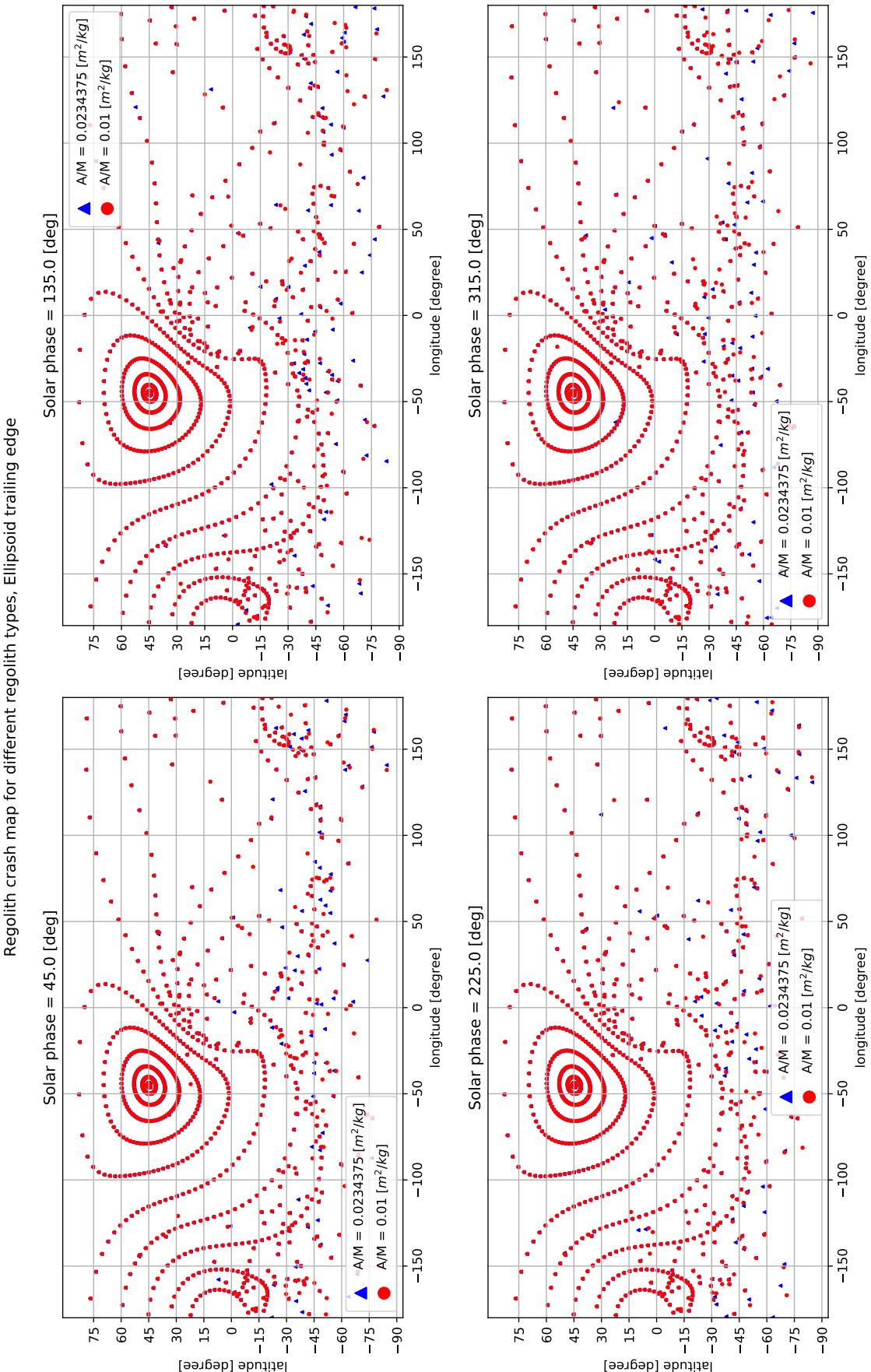


Figure 7.57: Re-impact map for regoliths of 3.2 g/cm^3 and 7.5 g/cm^3 but same particle radius of 1 cm. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

Regolith crash map for different regolith types, Ellipsoid trailing edge

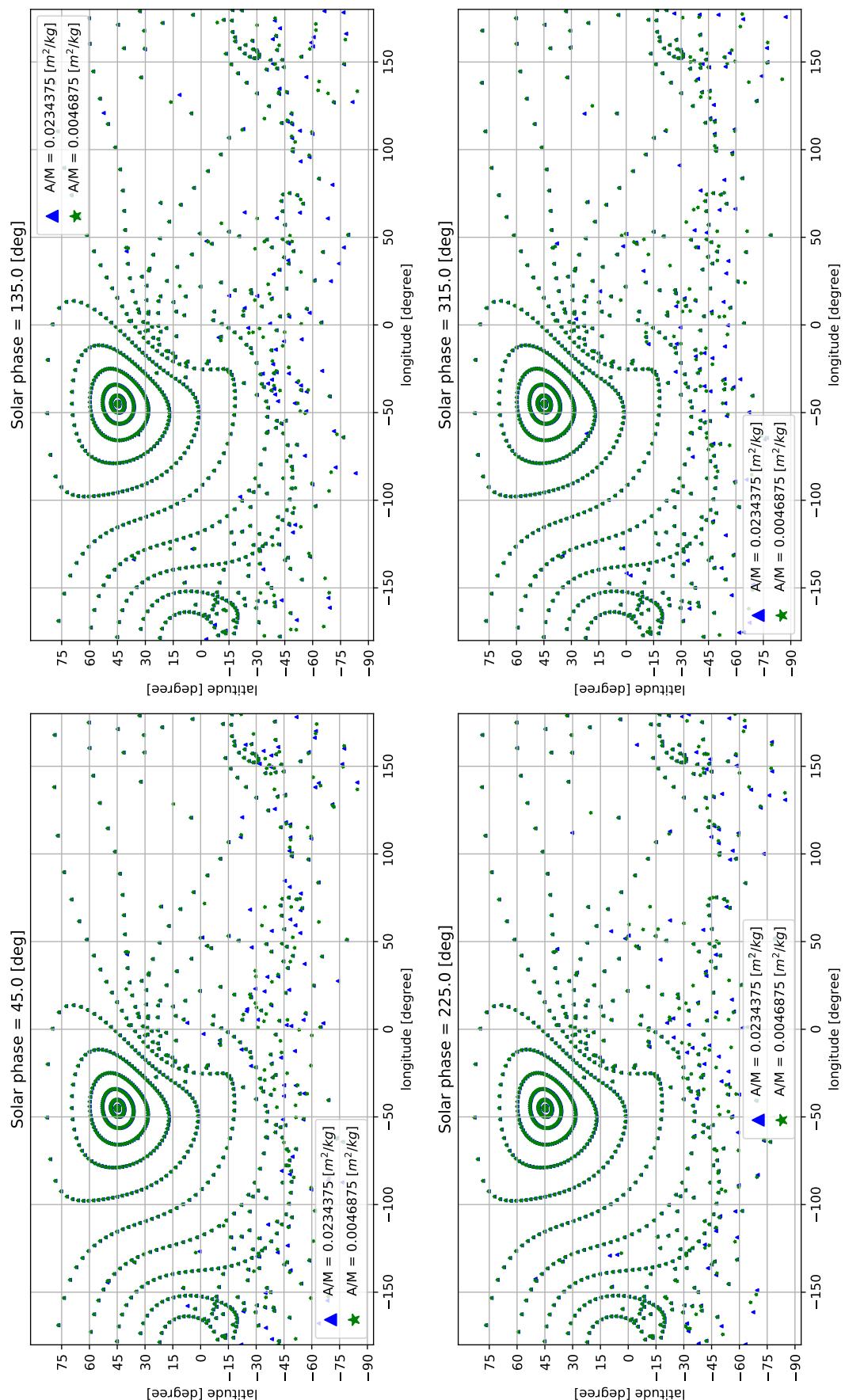


Figure 7.58: Re-impact map for regoliths of density 3.2 g/cm^3 and particle radii of 1 cm and 5 cm. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

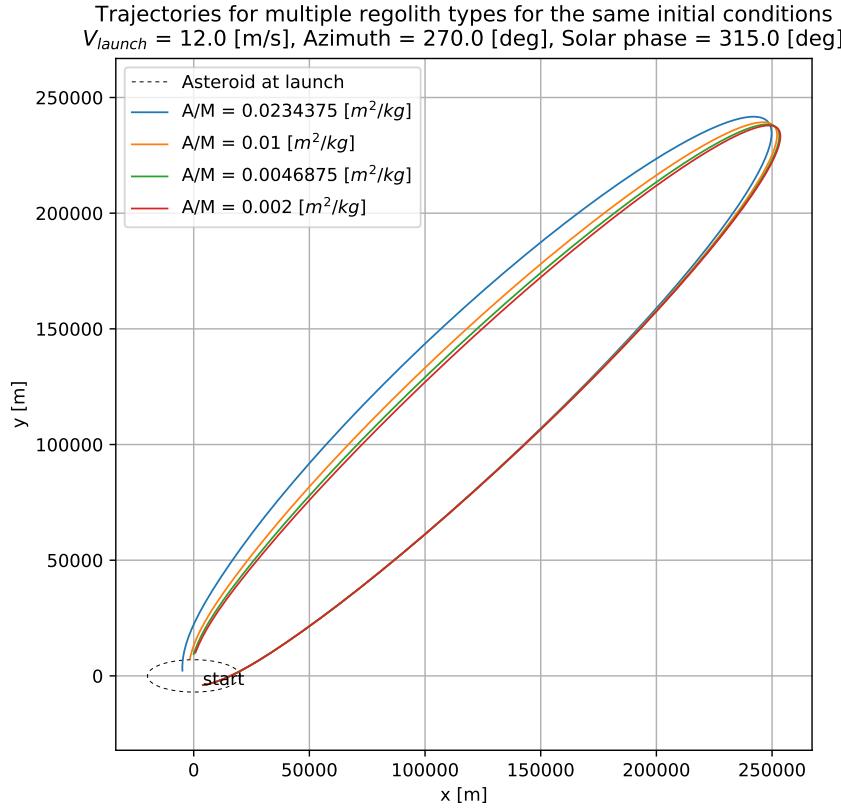


Figure 7.59: Re-impact trajectories shown for all regolith types, for launch velocity 12 m/s, azimuth 270° and initial Solar phase angle of 315°. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

Figure 7.60 shows the total number of escape cases. The numbers are fewer than those for the leading or the longest edge scenario since the bulk of the particles resorted to re-impact. Just like for the leading edge launch site, we will examine if the previously generalized relationship between the distribution of HEV data points and the escape behavior, is valid for a launch site in the trailing edge as well. Figure 7.61 shows the variation in HEV with the launch azimuth for three distinct velocities and for an initial Solar phase angle of 225°. The escape behavior as depicted for 13 m/s and 15 m/s is similar to that of the case for leading edge (therein for launch velocities 12 m/s and 14 m/s). As an example, for the launch velocity of 13 m/s in Figure 7.61, all HEV data points that don't correspond to a 0 HEV value and are uniformly distributed can be correlated to escaping from the asteroid without completing even a single revolution. However, just like with the case of the leading edge, these trajectories do not correspond to being hyperbolic from the instance of launch. For this example, the energy and eccentricity plot is shown in Figure 7.62 and the corresponding trajectories are shown in Figure 7.63. The specifics in these graphs are not important but one should look at the general trend for launch azimuths going from 200° to 325°. The explanation for the escape behavior at 15 m/s is the same as that covered for the leading edge case for 14 m/s and hence is not repeated here. For launch velocity 9 m/s in Figure 7.61, the HEV data points for particle LoGSP-1 are plotted from azimuth 270° to 285°. The energy and eccentricity plots for it is given in Figure C.25 and the trajectory evolution plots are given in Figure C.26. Unlike all the discussion we have had so far on this topic, we notice here that despite the distribution of HEV points, the particles take atleast one revolution around the asteroid before escaping.

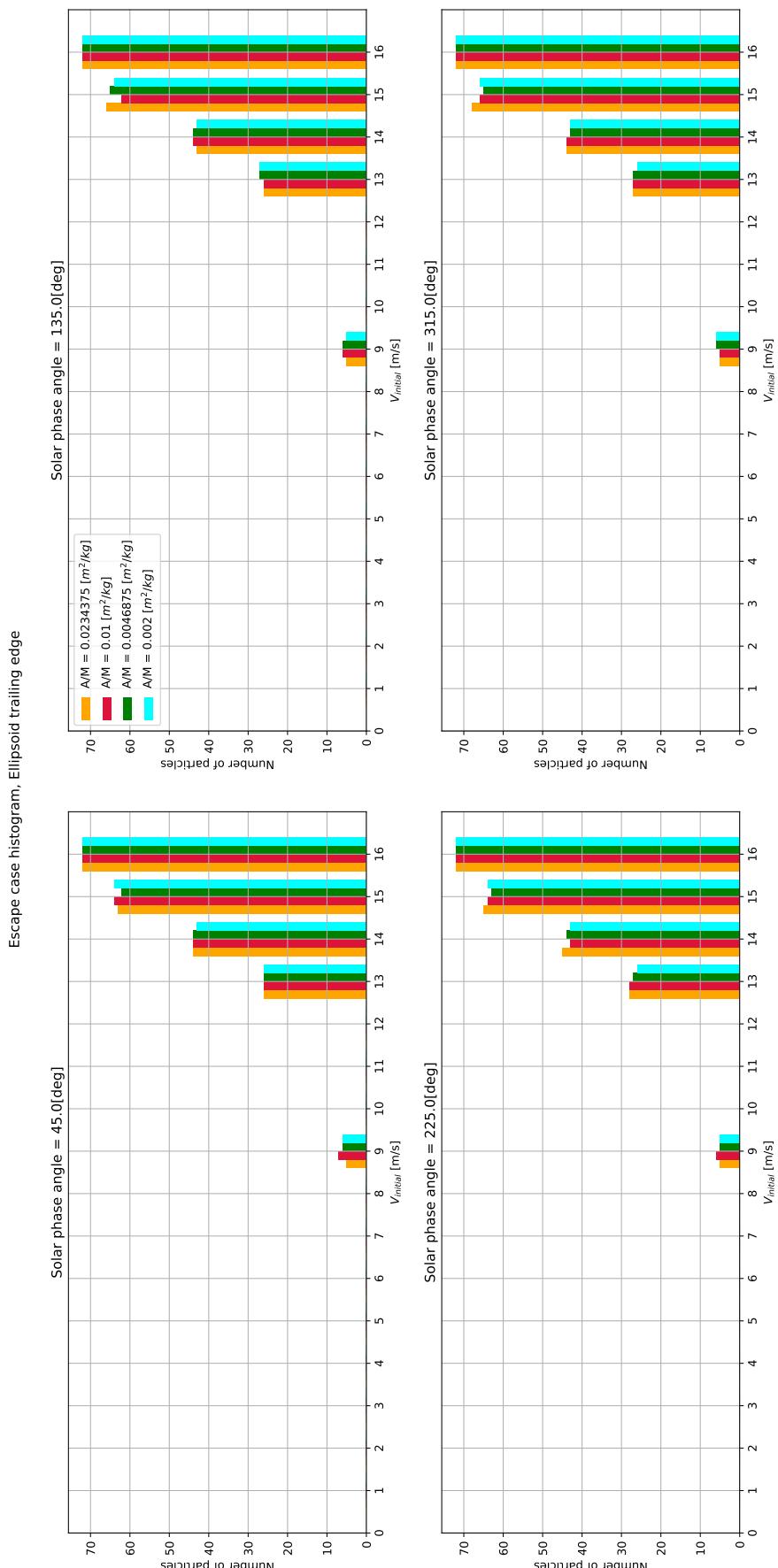


Figure 7.60: Number of all escape cases for all regolith types launched from the trailing edge of the asteroid. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

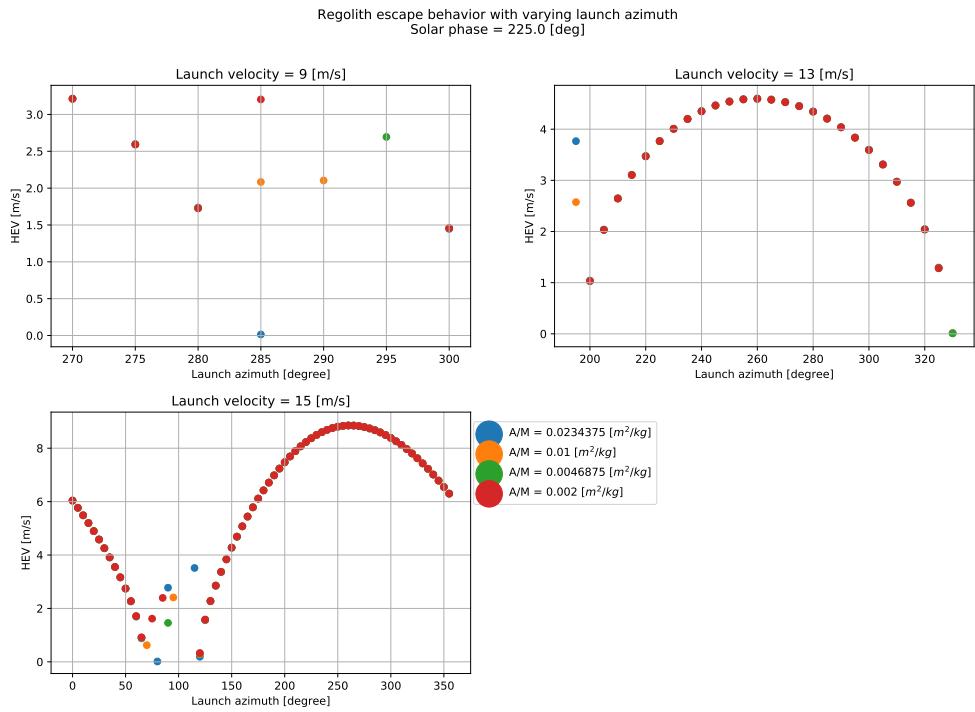


Figure 7.61: Plot depicting the variation in HEV with launch azimuth, for three distinct launch velocities and for all regolith types. The initial Solar phase angle is 225°. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

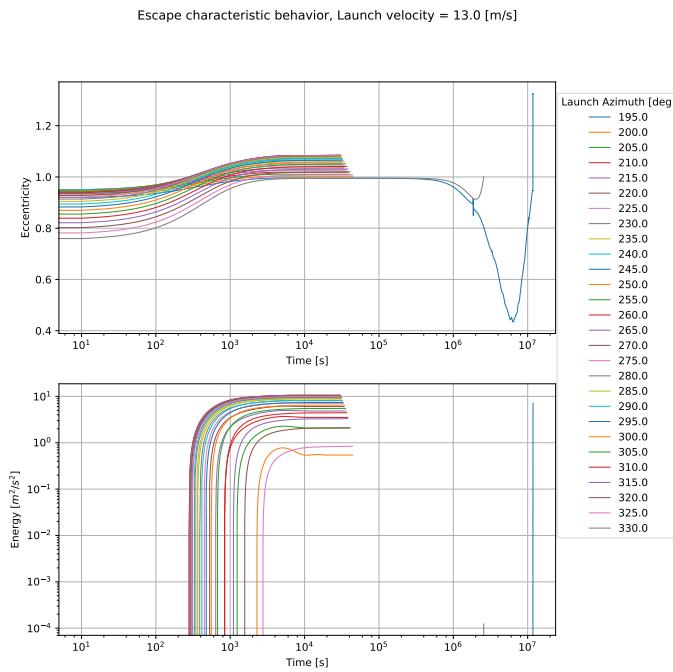


Figure 7.62: Plot depicting the variation in energy and eccentricity with time for escape cases, for particle LoGSP-1 launched with a velocity of 13 m/s, from Figure 7.61.

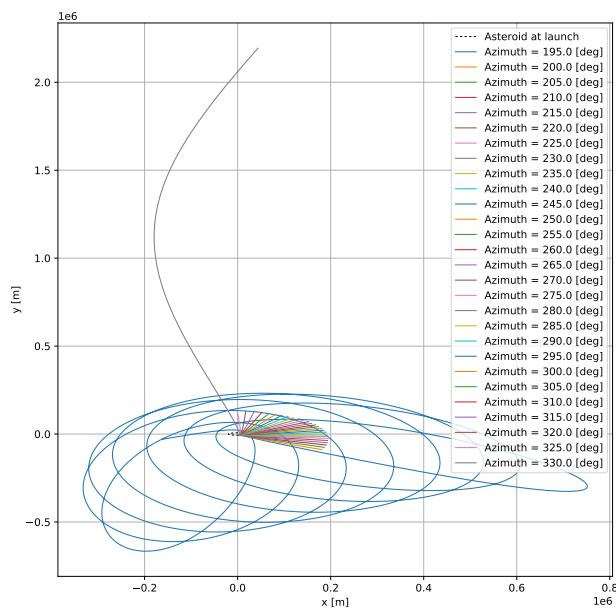


Figure 7.63: Plot depicting the trajectory evolution for escape cases, for particle LoGSP-1 launched with a velocity of 13 m/s, from Figure 7.61.

Figure 7.64 shows the total number of temporary capture cases for particles launched from the trailing edge of the asteroid. The numbers are far less than those for the leading and the longest edge launch cases. For the resolution of initial conditions employed in the simulation, atleast one capture case is obtained for all regolith types. Compared to the total particles launched, these numbers are extremely small. We'll look at an example of a capture trajectory for particle LoGSP-1 launched at 13 m/s and an initial Solar phase angle of 45°. The launch azimuth at which capture occurs is 330°. For the same initial conditions, we plot the trajectory for other regolith types as well for comparison. These are shown in Figure 7.65. While regolith LoGSP-2 re-impacts (orange curve), the remaining two particles are in temporary capture. Figure 7.66 shows the comparison between all capture trajectories for the given initial conditions.

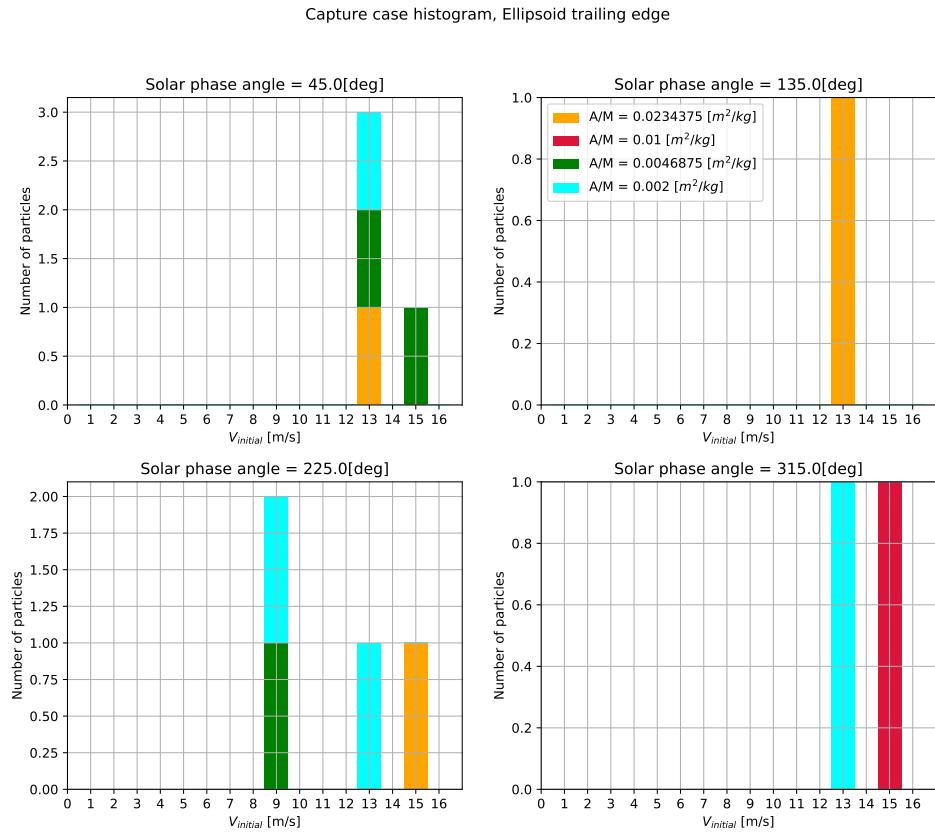


Figure 7.64: Number of all capture cases for all regolith types launched from the trailing edge of the asteroid. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

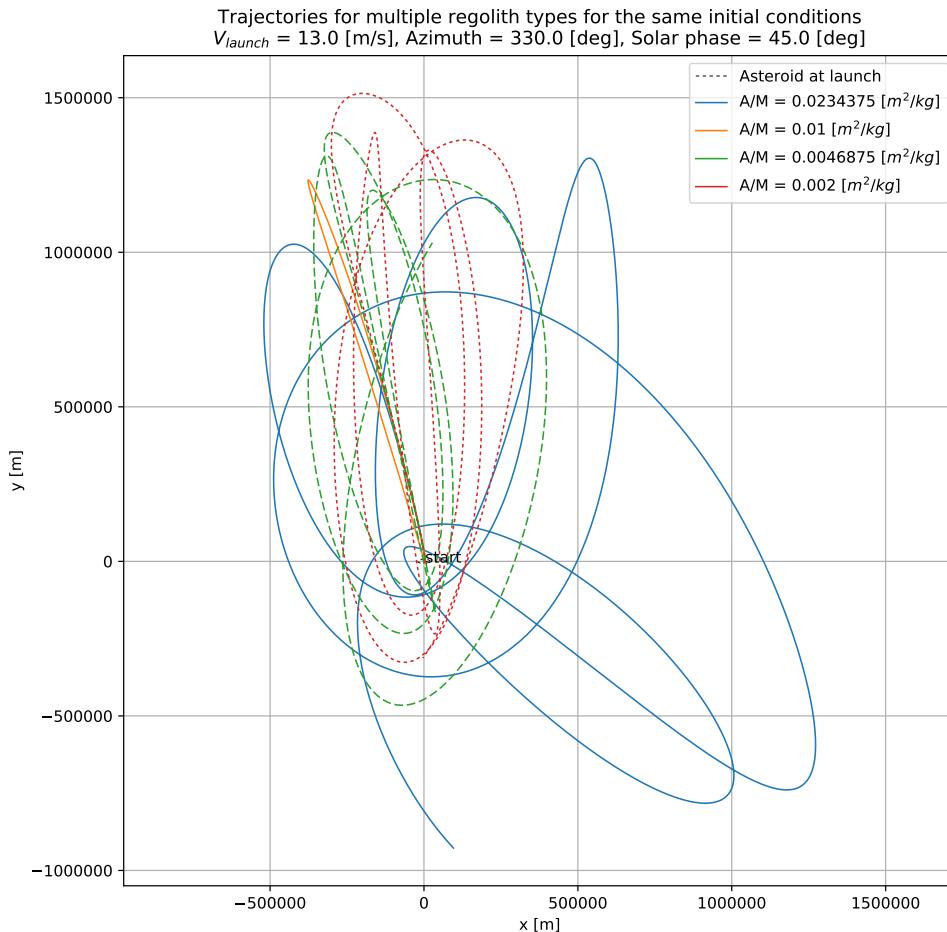


Figure 7.65: Capture trajectory example for particle LoGSP-1 launched from the trailing edge of the asteroid with initial velocity 13 m/s, azimuth 330° and initial Solar phase angle 45°. For the same initial conditions, the trajectories for other particles is plotted as well for comparison. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

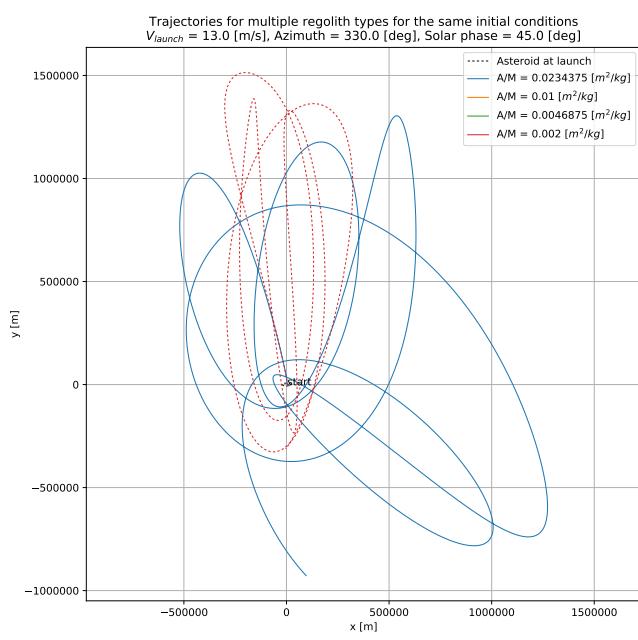
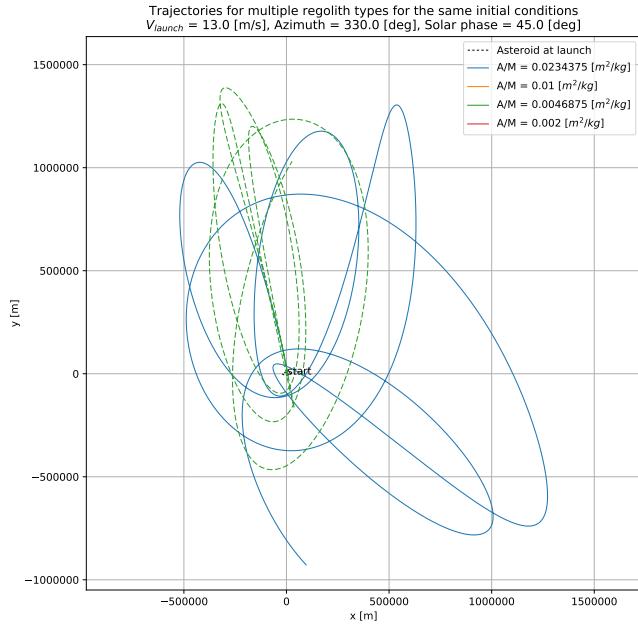


Figure 7.66: Comparing capture trajectories, from Figure 7.65, with each other for clarity. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

7.4.4 REGOLITH SORTING: APPLICATION TO SAMPLE SORTING AND MINING

In this section, we briefly discuss exploiting the motion of lofted regolith, under the presence of Solar perturbations, for the purposes of naturally sorting regoliths of different size and density. A natural sorting process like this is analogous to winnowing, which is done to separate grain from dust and junk on agricultural fields. This has applications to sorting material for sample collection for space exploration activities, and most importantly, for asteroid mining. For example, in our case, we have considered two different types of material, the first being Olivine which is basically metal silicates and the second being Fe-Ni metal alloy. For an asteroid mining company, the latter is more valuable than the former material type. If by exploiting natural motion of different types of regolith, under the presence of Solar perturbations, we can sort the desirable material from the undesired passively, then we wouldn't need extra equipment on a satellite or asteroid-rover to do the sorting actively.

Thus, we look at whether we can sort two different types of materials (and not all four regolith types as discussed before). These are regoliths LoGSP-1, which is Olivine material of density 3.2 g/cm^3 and grain radius 1 cm, and LoGSP-4, which is the Iron-Nickel alloy of density 7.5 g/cm^3 and grain radius 5 cm. Thus, assuming that both regolith types are homogeneously mixed with each other on the surface of the asteroid, we wish to see if SRP can be used to separate the larger metal alloy particles from the smaller silicate regolith. There are two ways in which these materials could potentially be sorted, both of which include lofting the particles artificially from a given launch site on the asteroid. The first method is to look at the re-impact locations for each particle type and see if majority of them have unique re-impact sites. In such a case, the desirable particle could be collected by pre-placement of storage capsules at those respective re-impact locations. The second method is in-orbit separation, wherein the particles eventually obtain different trajectories while in orbit and hence naturally separate themselves out. Placing a collection spacecraft in orbit at the right location can then collect the desired material in space.

We begin by discussing the on-surface sorting method. Just like before, we look at re-impact maps for particles launched from the longest (Figure 7.67), leading (Figure 7.68) and the trailing edges (Figure 7.69). In each scenario, majority of the particles have overlapping re-impact locations and only a small fraction appear to be passively sorted out. If the idea is to collect sorted material on the surface then in this case the method seems to be inefficient.

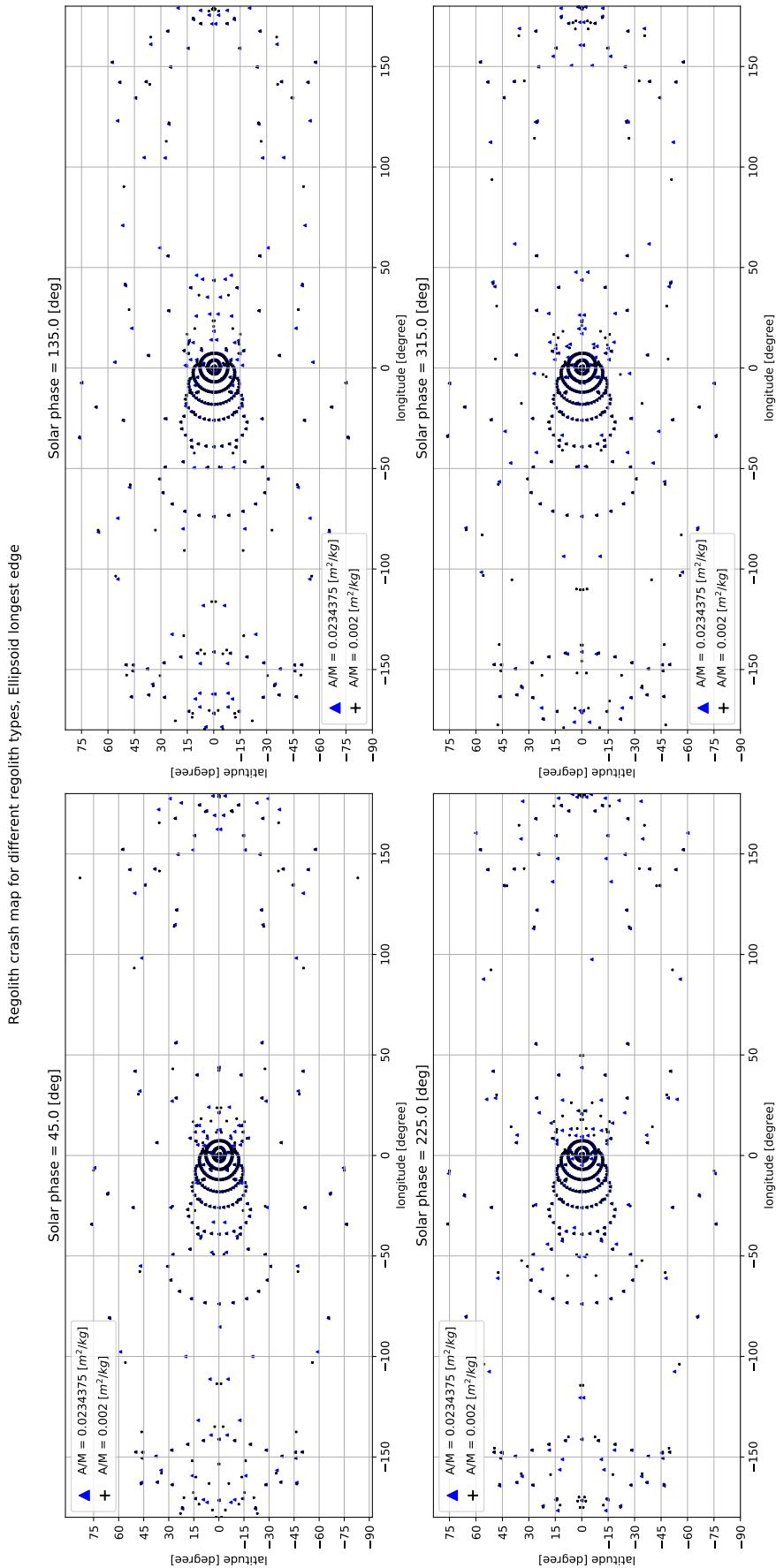


Figure 7.67: Re-impact map for regoliths LoGSP-1 and LoGSP-4. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

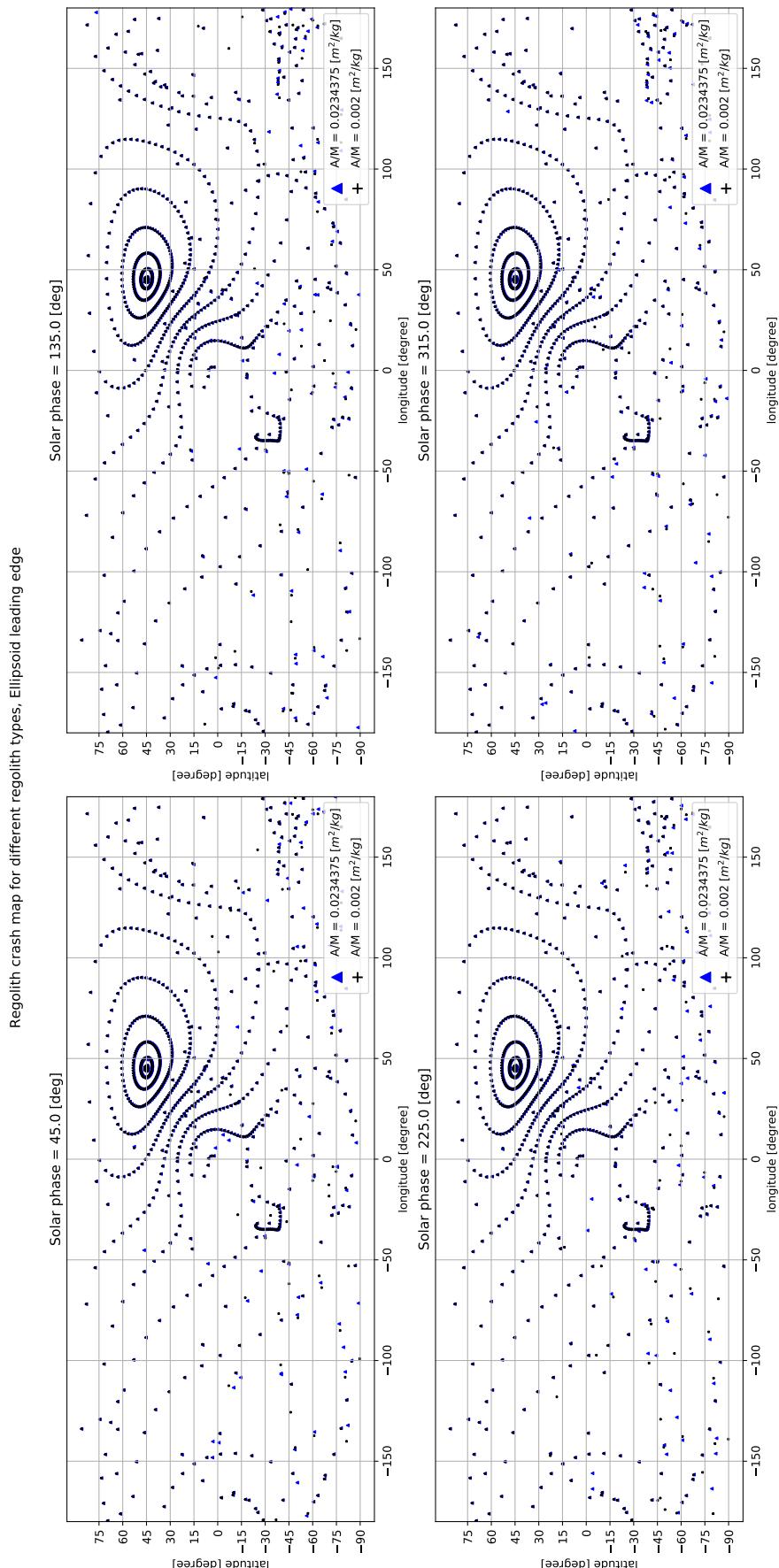


Figure 7.68: Re-impact map for regoliths LoGSP-1 and LoGSP-4. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

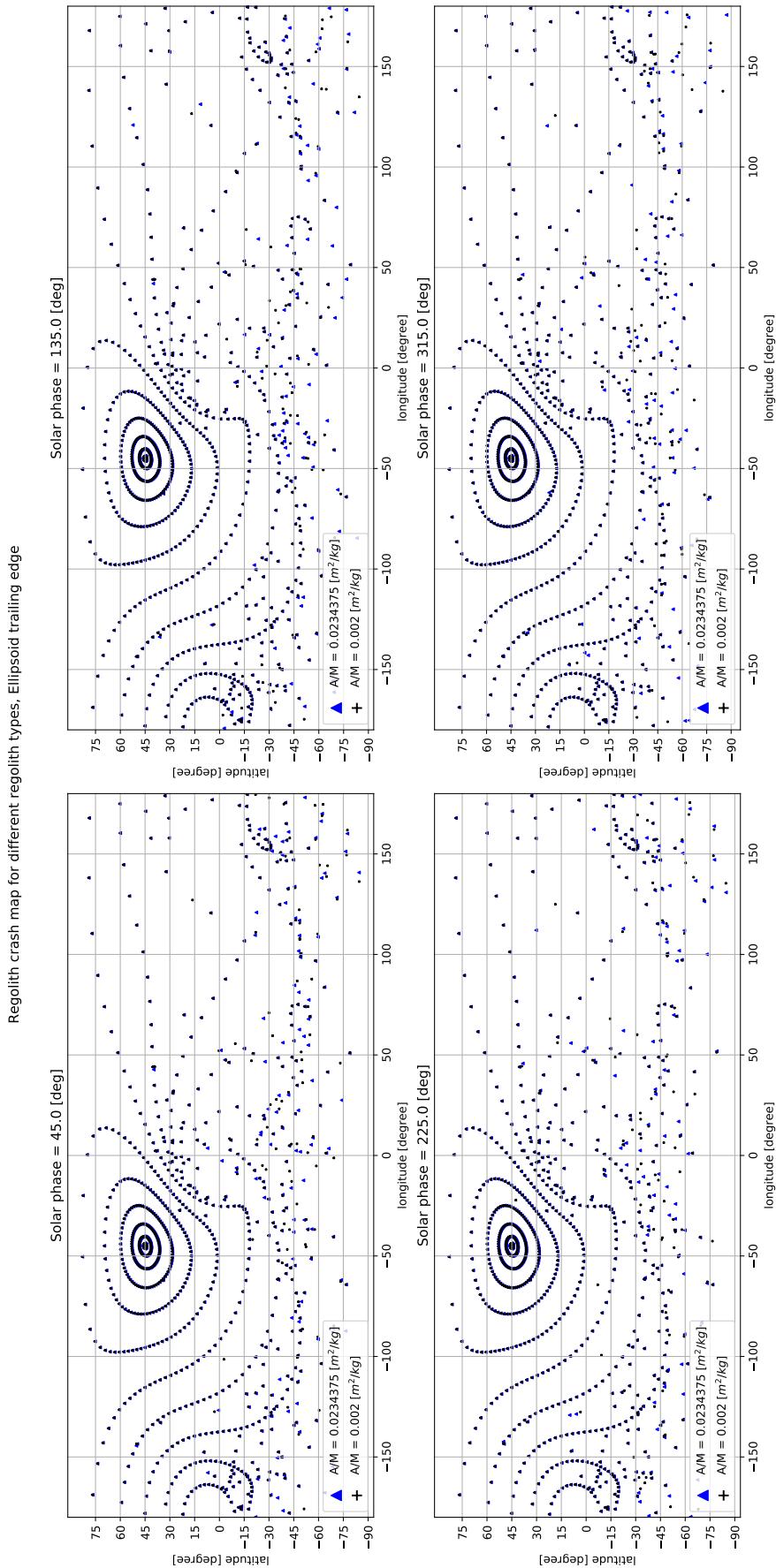
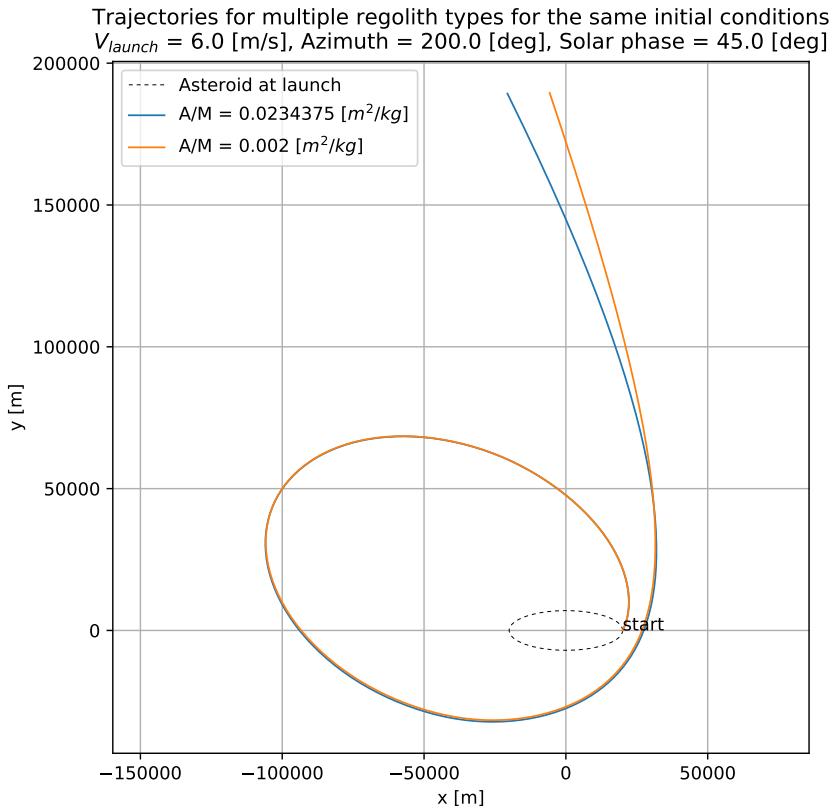
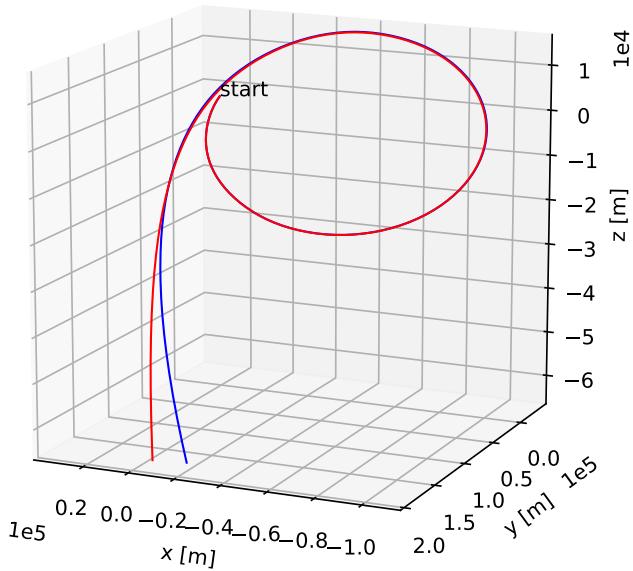


Figure 7.69: Re-impact map for regoliths LoGSP-1 and LoGSP-4. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

We now look at the in-orbit passive separation or sorting of the particles. We considered only those cases where the regoliths would spend atleast 24 hours in orbit post launch. This assumption was made to account for time that may be required by a spacecraft to re-orient and position itself before it can collect the sorted material. Figure C.27 shows the initial conditions for particles that result in more than 24 hours in orbit when launched from the longest edge. Figure C.28 and Figure C.29 show the same for leading and trailing edges respectively. We look at some example plots to see whether these separations are feasible enough in regards to material sorting or not. Figure 7.70 shows the scenario where both particles, launched from the longest edge, lead to an escape situation but obtain different trajectories before doing so. Similarly, Figure 7.71 shows an example of in-orbit particle separation, launched from the leading edge, where both eventually re-impact the surface. And finally for the trailing edge, Figure 7.72 shows the capture scenario for both the particles where the particles remain separated from each other till the end of simulation. All three of these examples tell dictate a much better possibility of separating the material in-orbit than on the surface through re-impact scenarios. However, the number of these cases where particles stay in orbit for a relatively longer time are much lesser than the total number of particles launched in the first place. Also once lofted, all particles would move forth in their respective trajectories simultaneously, and since the particles are going in all possible directions, it will be impractical for a spacecraft to be able to collect all the sorted material at once. Thus even the second method may not be suitable for in-orbit material collection unless an approach can be devised which keeps lofting the mixture of regolith in only a few particular directions.

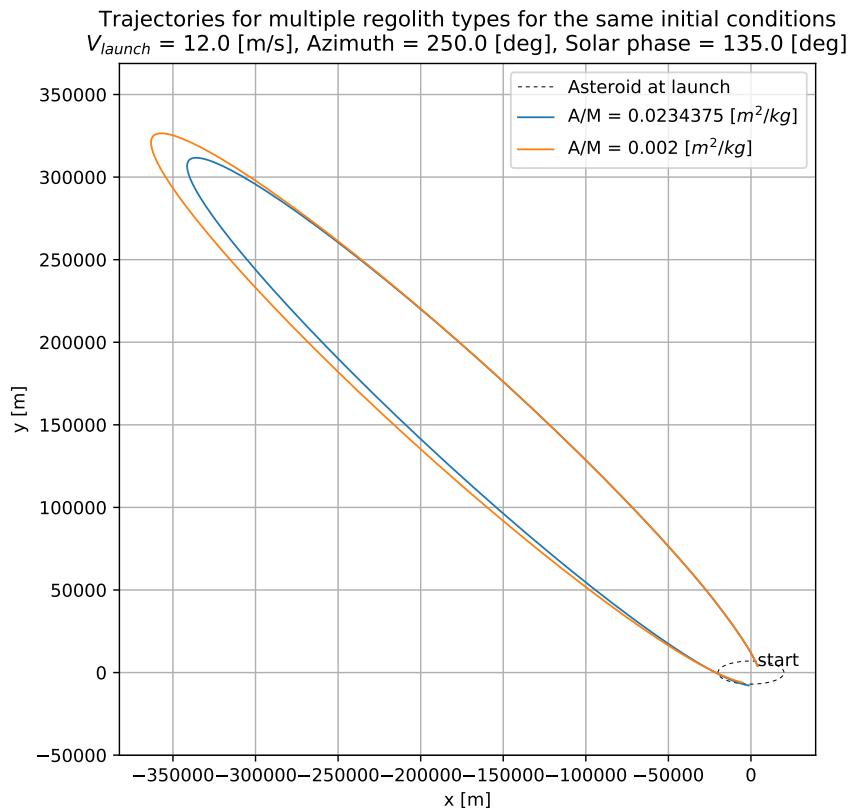


(a)

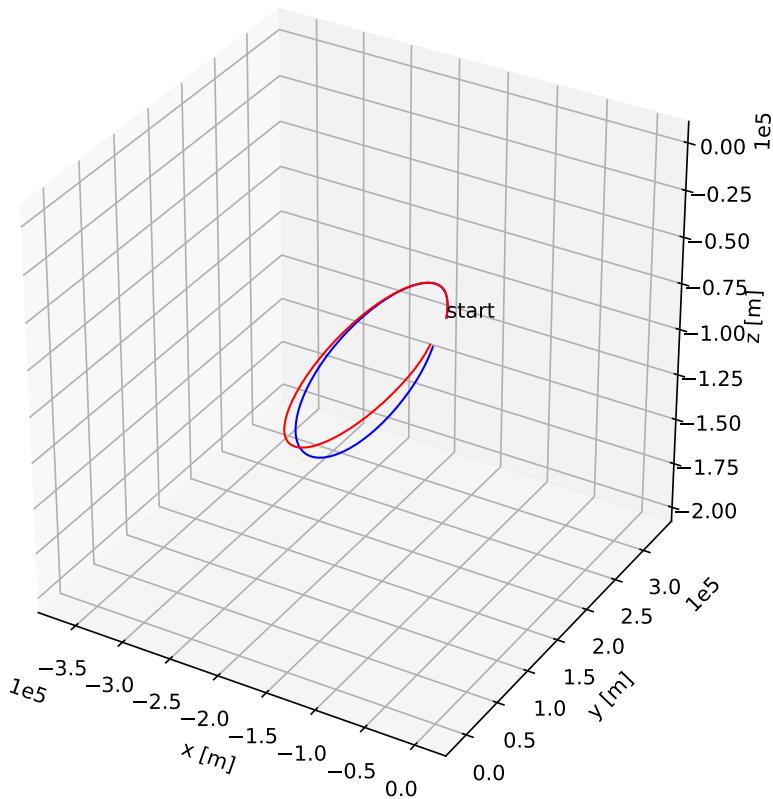


(b)

Figure 7.70: Trajectory plots, both 2D and 3D, for particles LoGSP-1 and LoGSP-4 launched from the longest edge of the asteroid that eventually lead to particle separation in orbit.

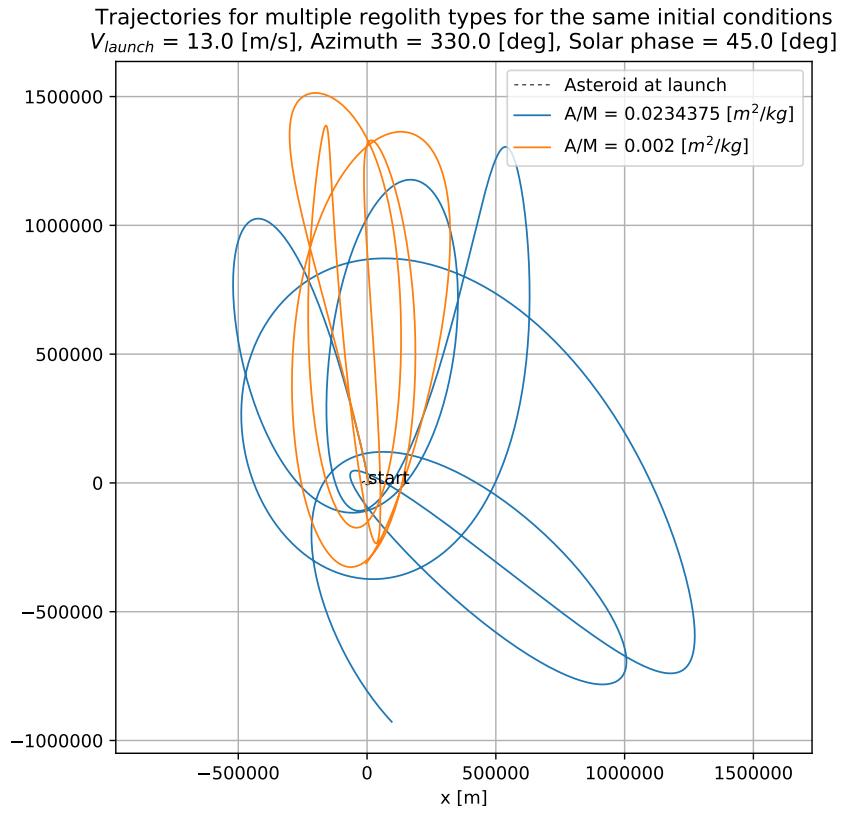


(a)

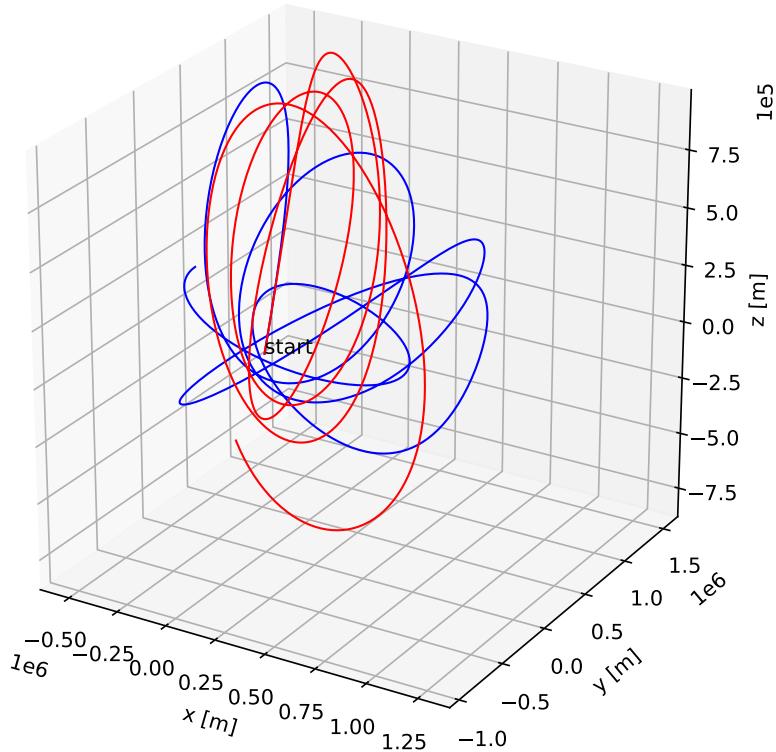


(b)

Figure 7.71: Trajectory plots, both 2D and 3D, for particles LoGSP-1 and LoGSP-4 launched from the leading edge of the asteroid that eventually lead to particle separation in orbit.



(a)



(b)

Figure 7.72: Trajectory plots, both 2D and 3D, for particles LoGSP-1 and LoGSP-4 launched from the trailing edge of the asteroid that eventually lead to particle separation in orbit.

Part IV

Conclusions and Recommendations

8

CONCLUSIONS

This chapter, finally, marks the end of the work done in this thesis. The work presented in this report was an effort to contributing towards understanding and characterizing the orbital motion of regolith and its final fate, once it was lofted from the surface of an asteroid. Much of the work done in the past involved mostly analytical methods to understand the orbital motion of the regolith, however, in this thesis we employed mainly a numerical simulation approach. For this, a simulator, called NAOS, was developed from scratch that can perform high fidelity simulation for an orbiting particle and with high accuracy when it comes to close proximity motion.

A constant density ellipsoid model was used to model the asteroid so as to account for a non-uniform gravity field without posing the known divergence problems of other gravity field models such as the spherical harmonics. The polyhedron model was not used either to decouple any irregular body shape effects on the motion of the regolith and correctly judge the influence of perturbations. The simulator also accounted for perturbations from the Sun, namely the SRP and the STBE. Every single particle lofted from the surface of the asteroid, was given a specific velocity and a direction relative to the local North direction and the local surface Normal. By this method we were able to create and launch multiple particles, shaping out like a cone, resembling how regolith would be lofted when an object impacts the asteroid's surface. For each regolith in this cone of particles, the simulator would then create a launch location position vector (from a given longitude and latitude) relative to the asteroid's centre and then launch it with the specified velocity and direction. We launched particles from three different launch locations, i.e. from the leading, trailing and longest edge of the ellipsoid shaped asteroid.

We'll now present a summary of all the results by providing answers to all the sub-research questions mention in Chapter 3.

- 1. Does the regolith, launched from different locations such as leading, trailing, longest edge of an asteroid, show characteristic differences with regard to its final fate?**

Yes, there were significant differences in terms of final fate between particles launched from the three distinct launch locations. In case of the longest edge, there were an almost equal number of escape and re-impact cases, and relative to those, very few temporary capture cases. For the leading edge, the majority of the particles resulted in re-impact (much more than that for the longest edge case) and only very few resulted in an escape. Even fewer particles attained temporary capture orbits. The largest number of re-impact cases were observed for particles launched from the trailing edge. Subsequently, it recorded the lowest escape and temporary capture cases.

A relationship was observed between the distribution of HEV data points and the number of orbital revolutions it took for a particle to escape. If the distribution of the points was uniform and continuous, then it was correlated to particles escaping without completing even a single orbital revolution. However, this correlation was found to be completely true for only the longest and leading edge cases. For the trailing edge scenario as well, this correlation could be established with all launch velocities that lead to an escape, but, one particular launch velocity case presented an exception. The correlation turned out to be false in case of this exception.

The re-impact maps for the leading edge case appeared to have the least amount of separation for different regolith types, followed by the trailing edge and then the longest edge.

2. Can we establish a non-conservative analytical expression to determine guarantee escape speed in presence of perturbations?

A new semi-analytical method was developed to determine a non-conservative guaranteed escape speed for the regolith as the conservative guaranteed escape speed method couldn't detect escape cases where the launch velocity was below it. The new method, however, did not work as expected and proved to be of no use in detecting any escape cases.

In addition to this, we also observed that the conventional guaranteed escape speed method could not be used to distinguish between cases that took one or more revolutions before escaping and those which took zero revolutions.

Note that this sub-research question was analyzed by conducting simulations only from the longest edge of the asteroid and in absence of Solar perturbations.

3. What causes the regolith to enter into a temporary capture orbit around the asteroid?

The answer to this question was found by comparing two particles launched with the same initial conditions, but one in the presence (perturbed scenario) and the other in the absence of Solar perturbations (unperturbed scenario). It was found that the perturbations would produce enough changes in the trajectory of a regolith such that its phase with respect to the asteroid would change relative to the unperturbed scenario. These changes, for example in terms of range to the particle, would slowly develop as cm level difference and eventually build up to km level differences. Because of these changes, for the same epoch, the particles from the perturbed and unperturbed scenario would have significantly different locations along the trajectory. Thus this (favorable)change in the *phase* of the particle with respect to the asteroid ultimately leads to a capture orbit.

Along with this, it was also discovered that both SRP and STBE are important in obtaining a capture orbit. This was inferred when for the existing capture cases, the simulation was re-run with the same initial conditions but by removing one of the perturbations each time. The effect of removing STBE was not as drastic as that of removing SRP, however both were needed to get the original capture orbit.

4. For the same launch conditions, how does the orbital behavior and final fate of the regolith differ for different particle sizes and densities?

For the same launch conditions, all regolith types don't necessarily need to have the same final outcome. There is no way in which we can predetermine their final fates either without running a simulation. That said, when we look at the trajectory plots for all regoliths launched with the same conditions, we note that the particles initially have overlapping trajectories, but eventually the trajectories start separating out from each other. The order of separation is determined by the area-to-mass ratio of the particle wherein the one with the maximum ratio separates out first. The eventual trajectory difference between the particles of successive area-to-mass ratios is also significantly large.

We witnessed certain cases where all regolith types eventually have the same final fate, when launched with the same initial conditions. If it was escape, they had different HEV and if it was re-impact, then they re-impacted the surface at different locations.

5. Can we exploit the orbital behavior of lofted regolith for sorting material of different sizes and densities as an application for asteroid mining?

The research done for this question was extremely rudimentary however we were able to derive some preliminary conclusions out of it. We compared orbital motion of regoliths of two different densities and sizes. One of the methods was to see if these particles would re-impact the surface on extremely different and far away locations such that they would be passively sorted. However, this was not found out to be true. For all three launch locations, majority of the particles had overlapping re-impact locations which rendered this method unfeasible.

The second method was to see if passive sorting can be achieved while the particles were in orbit. This method provided more success than the previous one in sorting out the regoliths. We observed only those cases where the regoliths would spend more than one Earth day in orbit in order to account for time taken by spacecraft to maneuver and position itself to collect the sorted material in orbit. However, the number of such cases was small and even within those, the method seemed impractical since the particle trajectories would go in all possible directions and collecting them all at once by a single spacecraft is impossible.

9

RECOMMENDATIONS FOR FUTURE WORK

...[TBD]...

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Appendices

A

OTHER GRAVITY POTENTIAL MODELS

This chapter briefly discusses a few other gravity potential models. These models were not used in this thesis but are still presented here as supplementary material.

A.1 SPHERICAL AND ELLIPSOIDAL HARMONICS

One of the most common methods for modeling the gravity potential of any celestial body is the *spherical harmonics* model. In that, a sphere whose radius is equal to the maximum dimension of the irregular body, circumscribes it and this sphere is called the *Brillouin sphere* (see Figure A.1). The spherical harmonics model then induces deformities on the Brillouin sphere, thereby producing a non-spherical gravity field. The spherical harmonics gravity potential is stated as follows (Scheeres 2016):

$$U(r, \delta, \lambda) = \frac{\mu}{r} \sum_{l=0}^{\infty} \sum_{m=0}^l \left(\frac{r_0}{r}\right)^l P_{lm}(\sin \delta) [C_{lm} \cos m\lambda + S_{lm} \sin m\lambda] \quad (\text{A.1})$$

where U is the gravity potential calculated at a distance r from the centre of the Brillouin sphere at latitude δ and longitude λ , μ is the gravitational parameter of the irregular body or asteroid, r_0 is the radius of the Brillouin sphere, P_{lm} are the associated Legendre functions, C_{lm} and S_{lm} are the spherical harmonics coefficients which account for shape and density variations (Romain et al. 2001), and l and m are the degree and order, respectively, of the spherical harmonic expansion. The definitions and calculations for the associated Legendre functions and the harmonics coefficients has been explained in detail by Scheeres 2016 and is not repeated here for brevity. The majority of gravity field perturbations can be accounted for by just considering the second degree and order in the spherical harmonics expansion. The potential is then expressed as follows (Scheeres 2016):

$$U = \frac{\mu}{r} \left[1 + \left(\frac{r_0}{r}\right)^2 \left\{ C_{20} \left(1 - \frac{3}{2} \cos^2 \delta\right) + 3C_{22} \cos^2 \delta \cos(2\lambda) \right\} \right] \quad (\text{A.2})$$

where the spherical harmonic coefficients can be obtained from the principle moments of inertia as defined in Scheeres 2016.

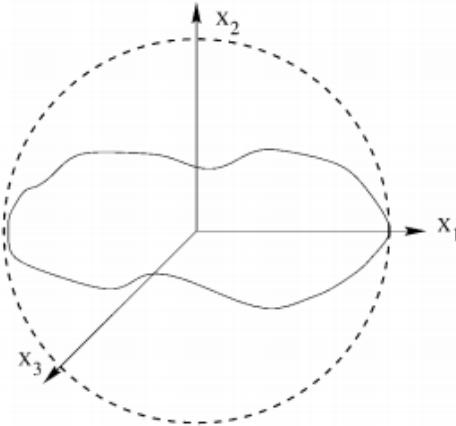


Figure A.1: Brillouin sphere or the circumscribing sphere around an irregular body (Romain et al. 2001).

Now consider a general statement for the gravity field of any arbitrary mass distribution \mathcal{B} (Scheeres 2016):

$$U = \frac{\mu}{V} \int_{\mathcal{B}} \frac{dV}{|\vec{r} - \vec{\rho}|} \quad (\text{A.3})$$

where V is the volume, \vec{r} is the position vector to the point where the potential is being calculated, $\vec{\rho}$ is the position vector to the discrete mass distribution within \mathcal{B} . If the potential is being calculated for a point that lies outside the maximum radius of the mass distribution being considered, then the integrand in Equation (A.3) can be expanded into the following Laplace series form (Scheeres 2016):

$$\frac{1}{|\vec{r} - \vec{\rho}|} = \frac{1}{r} \sum_{i=0}^{\infty} \left(\frac{\rho}{r} \right)^i P_{i0} \left(\frac{\vec{r} \cdot \vec{\rho}}{r\rho} \right) \quad (\text{A.4})$$

where P_{i0} are the Legendre polynomials. Thus using Equation (A.4), the integral in Equation (A.3) can be restated as follows (Scheeres 2016):

$$\frac{\mu}{r} \cdot \left[\frac{1}{V} \int_{\mathcal{B}} \left(\frac{\rho}{r} \right)^i P_{i0} \left(\frac{\vec{r} \cdot \vec{\rho}}{r\rho} \right) dV \right] \quad (\text{A.5})$$

There is a one-to-one correspondence between the term in the square brackets in Equation (A.5) and the i th degree and order spherical harmonics gravity field. Thus by looking at the Laplace series in Equation (A.4) and the integral in Equation (A.5), we can infer on the convergence or divergence of the spherical harmonics gravity field. Since the maximum radius of the mass distribution in case of the spherical harmonics model would be that of the circumscribing or Brillouin sphere, then for a point on this sphere, i.e. $r = |\vec{\rho}|$, the Laplace series is not defined and for a point inside the sphere, i.e. $r < |\vec{\rho}|$, the Laplace series diverges. This is the limitation for using the spherical harmonics model for an irregular body when one wants to compute orbital motion in close proximity to the body. If the computation points are within the Brillouin sphere, the spherical harmonics series might diverge to a value that does not represent the true gravity potential value and hence lead to errors in orbit computations. We can see from Figure A.1 that the volume of divergence for irregularly shaped asteroids can be quite significant. Thus, this model is definitely not suitable for our research since we are dealing with close-proximity orbits and above all, particle re-impact scenarios.

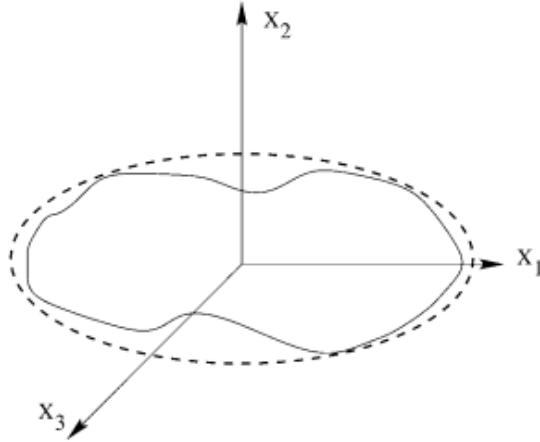


Figure A.2: Brillouin ellipsoid or the circumscribing ellipsoid around an irregular body (Romain et al. 2001).

The above problem can be mitigated, to a certain extent, by using the *ellipsoidal harmonics* expansion for representing the gravity potential of an irregular body. An extremely detailed account on this model is given by Dechambre et al. 2002. In the ellipsoidal harmonics model, instead of a sphere, a triaxial ellipsoid is used to circumscribe the irregular body and this proves to be a better fit as shown in Figure A.2. The ellipsoidal harmonics potential is given as follows (Dechambre et al. 2002):

$$U(\lambda_1, \lambda_2, \lambda_3) = \mu \sum_{n=0}^{\infty} \sum_{p=1}^{2n+1} \alpha_{np} \frac{E_n^p(\lambda_1)}{E_n^p(\lambda_1^{ref})} \times E_n^p(\lambda_2) E_n^p(\lambda_3); \lambda_1 \leq \lambda_1^{ref} \quad (\text{A.6})$$

$$U(\lambda_1, \lambda_2, \lambda_3) = \mu \sum_{n=0}^{\infty} \sum_{p=1}^{2n+1} \alpha_{np} \frac{F_n^p(\lambda_1)}{F_n^p(\lambda_1^{ref})} \times E_n^p(\lambda_2) E_n^p(\lambda_3); \lambda_1 \geq \lambda_1^{ref} \quad (\text{A.7})$$

where \$(\lambda_1, \lambda_2, \lambda_3)\$ are the ellipsoidal coordinates, which are basically three real roots (solutions) in terms of \$s\$ for the following conic equation (Garmier et al. 2002):

$$\frac{x^2}{s^2 + a^2} + \frac{y^2}{s^2 + b^2} + \frac{z^2}{s^2 + c^2} = 1 \quad (\text{A.8})$$

where \$(x, y, z)\$ are the Cartesian coordinates and \$(a, b, c)\$ are the semi-axes of the reference ellipsoid circumscribing the irregular body (note that \$a = \lambda_1^{ref}\$). In Equations (A.6) and (A.7), \$(\lambda_1, \lambda_2, \lambda_3)\$ are analogous to the radius \$r\$, latitude \$\delta\$ and longitude \$\lambda\$, respectively, of Equation (A.1); \$\alpha_{np}\$ is the ellipsoidal harmonics expansion coefficient similar to the spherical harmonics coefficient \$C_{lm}\$ and \$S_{lm}\$; \$F_n^p\$ are the Lamé function of second kind of degree \$n\$ and order \$p\$ and is analogous to the attenuation term \$(r_0/r)^l\$ of the spherical harmonics expansion in Equation (A.1); \$E_n^p\$ is the Lamé function of the first kind of degree \$n\$ and order \$p\$; and finally, the product term \$E_n^p(\lambda_2) E_n^p(\lambda_3)\$ is analogous to the product term \$P_{lm}(\sin \delta)[C_{lm} \cos m\lambda + S_{lm} \sin m\lambda]\$ which in both cases models the surface harmonic (Garmier et al. 2002). A detailed description on definition and calculation of the ellipsoidal harmonic coefficients and the Lamé functions of the first and the second kind can be found in Dechambre et al. 2002 and is not repeated here for brevity.

Even in the case of ellipsoidal harmonics expansion model, the gravity potential calculated for a point inside the circumscribing ellipsoid can diverge from the true potential. But the advantage of this model over the spherical harmonics expansion is that the circumscribing reference ellipsoid reduces the volume of divergence around the irregular body, relative to a sphere, making close-proximity evaluations possible. However, relative to the spherical harmonics expansion, the computation of the basis functions for ellipsoidal harmonics, i.e. the Lamé functions of the first and the

second kind, is extremely complex. On top of that, with increasing degree of the harmonics model, the order of magnitude of the Lamé functions increases, which then runs the risk of arithmetic overflow, thereby impeding accurate calculations of the harmonic expansion for degrees above 10 to 15 (Reimond et al. 2016). However, in their research, Reimond et al. 2016 have devised a new method to calculate the basis functions using logarithmic expressions which allows accurate harmonic expansions for degrees of upto 500 but the computational complexity also increases significantly as stated by them. Ultimately, since we wish to express the motion of particles close to the surface of the asteroid, which also involves surface interactions, the approach of ellipsoidal harmonics expansion also fails.

A.2 CONSTANT-DENSITY POLYHEDRON

The gravity potential modeling methods discussed so far involved the use of surface harmonics on the circumscribing object (sphere or ellipsoid) to simulate a non-homogeneous gravity field for an irregular body. The major drawback with the harmonics approach was its divergence from the true potential value within the circumscribing volume. This problem can be mitigated all together by assuming a specific shape and density distribution for the irregular body in question. In this realm, there are the CDE and constant-density polyhedron gravity potential models. Unlike the harmonics expansion approach, these potential models are valid upto and on the surface of the shape that has been assumed for the irregular body in question (Scheeres 2016). Hence, these models are perfect to study the close-proximity motion of particles or spacecraft around an asteroid.

The irregular shape of an asteroid can be best represented by a polyhedron shape model (henceforth polyhedron) as shown in Figure A.3. Surface irregularities in the form of craters, large boulders, mountains etc. can be easily modeled with this method. A polyhedron is basically a 3D body that consists of several *vertices* which form triangular faces or *facets* that are connected to each other through the *edges* of each face. A triangular facet thus comprises of three vertices, three edges and a surface normal as shown in Figure A.4 (Scheeres 2016). A detailed derivation for the polyhedron gravity potential model is given in Werner et al. 1996 and concisely presented in Scheeres 2016 as well. Hence, we will only present a summary of this method in this section.

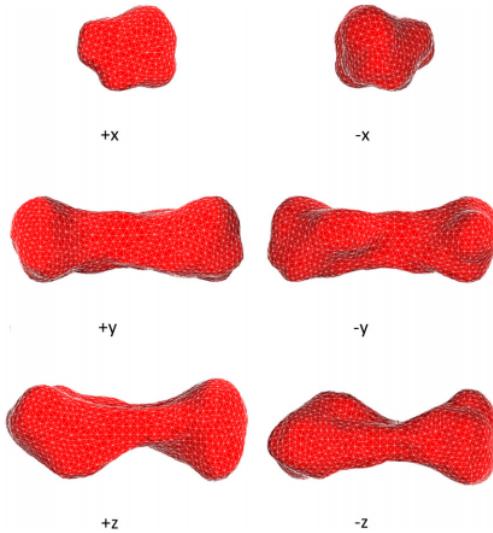


Figure A.3: Polyhedron shape model estimated for asteroid *Kleopatra* and shown in $\pm x, \pm y, \pm z$ axis directions. Constant-density has been assumed. Surface irregularities are easily modeled by this method (Yu et al. 2012).

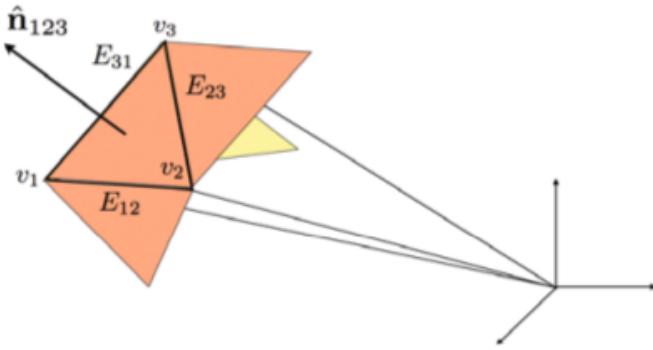


Figure A.4: Single facet of a polyhedron model depicting three vertices, three edges and a surface normal, associated with each facet in general (Scheeres 2016).

Each face or facet ' f' of the polyhedron is associated with three vertex vectors given as \vec{r}_i^f , where $i = 1, 2, 3$, and a unit normal vector $\hat{\mathbf{n}}_f$. The vector \vec{r}_i^f goes from each vertex of a facet to the field point \mathbf{P} where the potential has to be calculated. Each edge ' e' is associated to two vertex vectors \vec{r}_i^e , for $i = 1, 2$, and this edge connects two adjacent faces f and f' . Again, the vector \vec{r}_i^e goes from the edge vertices to the field point \mathbf{P} . The edge normal, corresponding to facet f , is denoted as $\hat{\mathbf{n}}_e^f$ such that it is perpendicular to the edge and the facet normal $\hat{\mathbf{n}}_f$ and is pointing away from the centre of the facet. For the same edge shared by facet f' , the edge normal $\hat{\mathbf{n}}_e^{f'}$ points in a different direction than $\hat{\mathbf{n}}_e^f$ and may not be parallel to it. With these definitions, the general formula for the polyhedron gravity potential is given as follows (Scheeres 2016):

$$U(\vec{r}) = \frac{\mathcal{G}\sigma}{2} \left[\sum_{e \in \text{edges}} \vec{r}_e \cdot \mathbf{E}_e \cdot \vec{r}_e L_e - \sum_{f \in \text{faces}} \vec{r}_f \cdot \mathbf{F}_f \cdot \vec{r}_f \omega_f \right] \quad (\text{A.9})$$

$$\mathbf{E}_e = \hat{\mathbf{n}}_f \hat{\mathbf{n}}_e^f + \hat{\mathbf{n}}_{f'} \hat{\mathbf{n}}_e^{f'} \quad (\text{A.10})$$

$$\mathbf{F}_f = \hat{\mathbf{n}}_f \hat{\mathbf{n}}_f \quad (\text{A.11})$$

$$L_e = \ln \left(\frac{r_1^e + r_2^e + e_e}{r_1^e + r_2^e - e_e} \right) \quad (\text{A.12})$$

$$e_e = |\vec{r}_1^e - \vec{r}_2^e| \quad (\text{A.13})$$

$$\omega_f = 2 \tan^{-1} \left(\frac{\vec{r}_1^f \cdot (\vec{r}_2^f \times \vec{r}_3^f)}{r_1^f r_2^f r_3^f + r_1^f (\vec{r}_2^f \cdot \vec{r}_3^f) + r_2^f (\vec{r}_3^f \cdot \vec{r}_1^f) + r_3^f (\vec{r}_1^f \cdot \vec{r}_2^f)} \right) \quad (\text{A.14})$$

where \vec{r} is the position vector to the field point \mathbf{P} from the origin of an asteroid-fixed reference frame; \mathcal{G} is the universal gravitational constant and σ is the density of the body being modeled; \vec{r}_e is the vector from any point along the edge ' e' to \vec{r} or the field point \mathbf{P} and in the same way \vec{r}_f denotes a vector from any point on the facet f to \vec{r} (Scheeres 2016); the term ω_f represents the solid angle subtended by a facet when viewed from the field point \mathbf{P} , or alternately, it is the angle subtended by the facet of the polyhedron on to a unit sphere centered at the field point \mathbf{P} ; L_e is analogous to the potential of a 1D straight '*wire*' and is computed for all facet edges in a polyhedron (Werner et al. 1996); \mathbf{E}_e is the edge dyad and is expressed as the sum of two outer products, forming a 3x3 matrix; and finally \mathbf{F}_f is the facet dyad which is simply the outer product of the facet normal vector with itself (Werner et al. 1996).

B

VERIFICATION & VALIDATION

In this chapter, we will present the results on verification and validation of the simulator that propagates the trajectory of a particle around an asteroid. The simulator, termed as NAOS, was developed independently by the author for this thesis and can be found at <https://github.com/agrawalabhishek/NAOS>. We verify the gravity model, the launch conditions for the regolith, the numerical integrator along with the equations of motion for the regolith, and finally the Solar perturbation models. We validate our simulator to ensure that our inferences for any scientific result out of it remains true and valid as well.

B.1 CONSTANT DENSITY ELLIPSOID GRAVITY MODEL

The CDE gravity model was tested for a singular target point by comparing the gravitational potential and acceleration values at that point as computed by NAOS, with external data obtained from another researcher at CSML¹. The parameters used for the test are given in Table B.1.

Table B.1: Parametric values used for testing the CDE gravitational potential model.

Parameter	Value	Units
Gravitational Parameter	446382.0	m^3/s^2
Alpha (longest axis of CDE)	20000	m
Beta (Intermediate axis of CDE)	7000	m
Gamma (Shortest axis of CDE)	7000	m
Target point x-coordinate	10000	m
Target point y-coordinate	13000	m
Target point z-coordinate	8000	m

The test values for the gravitational potential and the acceleration values at the specified target point are given in Table B.2.

¹Centre for Spaceflight Mechanics Laboratory; The thesis work was partly done at CSML, which is a research lab headed by Dr. Daniel Scheeres and is part of the University of Colorado, Boulder, USA.

Table B.2: Test values for verification of the CDE gravity model.

Parameter	Value		Units
Gravitational Potential	23.710052554396402		m ² /s ²
Gravitational Acceleration	x	-0.00044762916738340803	m/s ²
	y	-0.0009623388813999501	m/s ²
	z	-0.000592208542399969	m/s ²

The values obtained from the simulator NAOS were compared with the ones in Table B.2 upto the 12th decimal point and they matched, validating the gravity model implemented in NAOS.

B.2 REGOLITH LAUNCH CONDITIONS

In this section, we'll presents results on verifying whether the initial state vector for the regolith or the launch condition match to what is desired by the user. Here, an internal validation is done to ensure that the launch conditions registered in the output database match the raw value input to NAOS. We present graphical and numerical results in this regard.

B.2.1 LAUNCH LOCATION

The position vector to the launch location, from the centre of the ARF, is formed by giving only the launch location latitude and longitude as input. Thus, we have to verify whether the position vector conforms to a given angular input. We take the initial state vector, from the output databases of the dynamics simulator, of regolith launched from a few launch locations on the asteroid and convert the Cartesian coordinates back to latitude and longitude to see if the position vector was formed correctly. For the same Cartesian coordinates, we use the triaxial ellipsoid equation, and see if they solve the equation, to verify that the launch point lies on the surface of the asteroid.

We took 5 test locations from where regolith was launched and separately calculated the corresponding longitude and latitude for the launch location to check if the position vector was correctly formed. The position coordinates and the back-calculated latitude and longitude angles are shown in Table B.3.

Table B.3: Position vector to different launch locations and the corresponding Latitude and Longitude angles.

X [m]	Y [m]	Z [m]	Longitude [deg]	Latitude [deg]
20000.0	0.0	0.0	0.0	0.0
0.0	7000.0	0.0	90.0	0.0
0.0	0.0	7000.0	0.0	90.0
3316.14545023	1914.57746836	6632.29090046	30.0	60.0
-3961.38284482	-3961.38284482	5602.2413449	-135.0 (225.0)	45.0

The longitude and latitude values given in Table B.3 were calculated from the Cartesian coordinates and match those given as inputs to the dynamics simulator. The position vectors from Table B.3 with the CDE model are also depicted in Figure B.1.

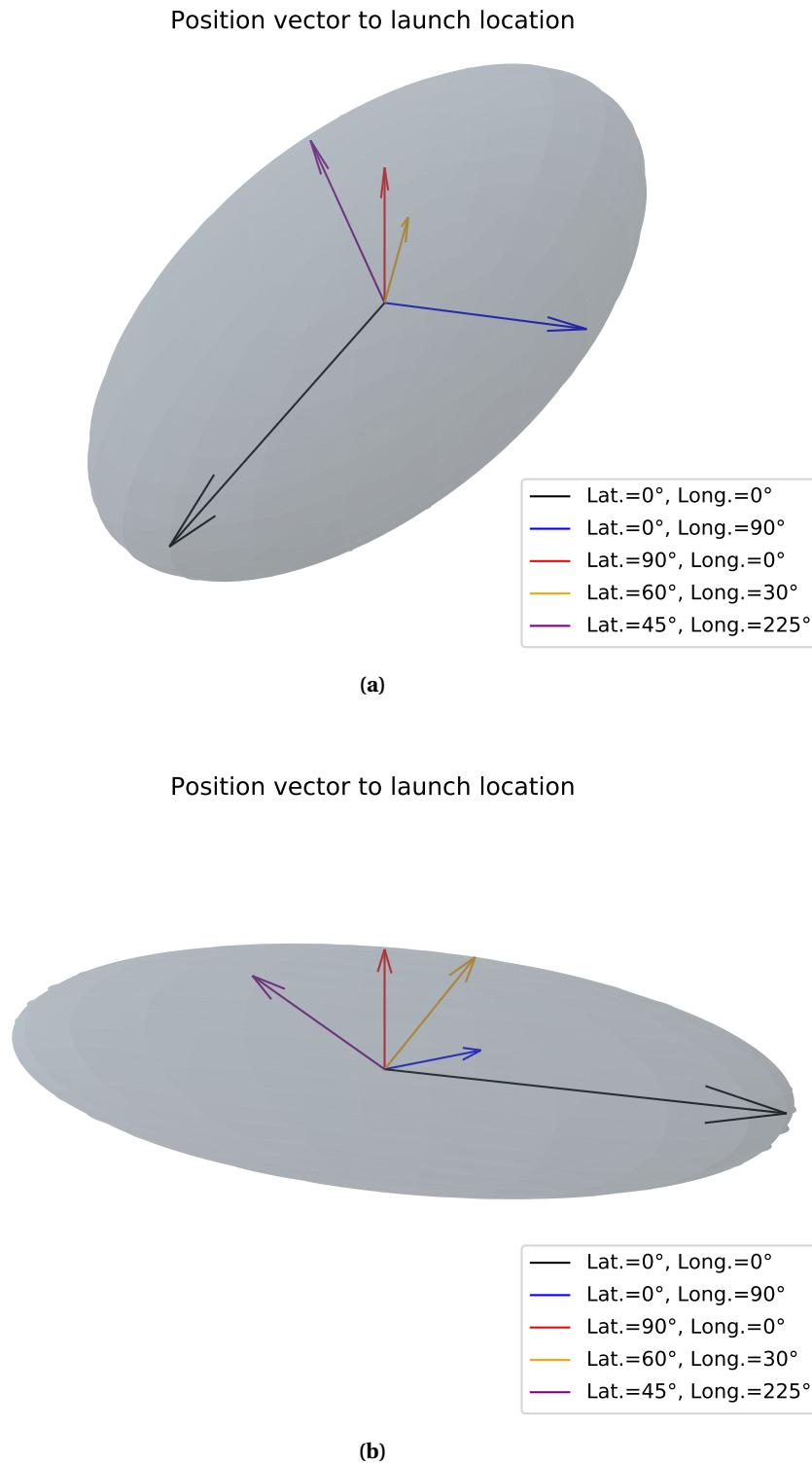


Figure B.1: Position vectors from Table B.3 shown with the CDE asteroid model.

We also checked whether the launch location in fact did lie exactly on the surface of the CDE asteroid and not inside or above the surface. This is really important because a wrong initial condition could produce erroneous results. Once we substitute the Cartesian coordinates into the triaxial ellipsoid equation (see Equation (4.57)) and if the output equals zero (when we take the term 1.0 on the right hand side in Equation (4.57)), then the launch point lies on the asteroid. We did an external validation for this but an internal validation always takes place within the dynamics simulator at all times whenever a new simulation is run.

Table B.4: Output of the triaxial ellipsoid equation for a few test launch location position vectors. An output of 0.0 means that the position vector is a perfect root of the equation and hence the launch location indeed lies on the surface of the asteroid.

X [m]	Y [m]	Z [m]	Output of triaxial ellipsoid equation
20000.0	0.0	0.0	0.0
0.0	7000.0	0.0	0.0
0.0	0.0	7000.0	0.0
3316.14545023	1914.57746836	6632.29090046	2.22044604925e-16
-3961.38284482	-3961.38284482	5602.2413449	0.0

B.2.2 LAUNCH VELOCITY

The launch velocity vector is formed by providing two angles and a magnitude. As explained in Section 4.6.2, the two angles are the declination angle from the normal vector at the launch location and the azimuth angle from the local North direction. The local North direction and the normal vector, in essence, act as the basis vectors for a local surface frame of reference which in turn help in the mathematical formulation of the velocity vector of Cartesian components. Since we are dealing with an ellipsoid and not a sphere, defining local North is not a straight-forward process as explained earlier in Section 4.6.2.

Thus, we first begin with verifying our methodology of forming the local surface frame, and ensure that the local Normal vector and the local North pointing vector are correctly formed and their lies no discrepancy. Then we verify whether the velocity vector, in terms of its Cartesian components, indeed makes the same angle with the surface frame as specified at the input of the simulation. The process for the second verification item is different from how the velocity vector was formed in the first place and hence the verification itself won't be a tampered or faulty process.

Let's first begin with a test launch location at the longest edge of the CDE. A graphical depiction of the frame is shown in Figure 4.7. This is the simplest location to perform any simulation or test because the surface normal vector here will be along the same direction as the position vector to the launch point from the centre of the ellipsoid. This also makes the test in itself a trivial task. Note that the surface normal is also the z-axis for the surface frame at the launch point. So for the current test location, if the cross product of the normal and the position vector is a zero vector, then the normal vector is verified. Table B.5 shows the result for this. Following this, if the angle between the normal vector and the x-axis basis vector for the surface frame is 90° , then the latter is pointing to the local North direction. This is because for the current test location, any vector perpendicular to the surface normal (or effectively to the launch location position vector) will be along the 0° meridian line crossing the test location. A simple dot product (again shown in Table B.5) between the x-axis basis vector of the surface frame and the normal vector will tell us if they are perpendicular to each other or not.

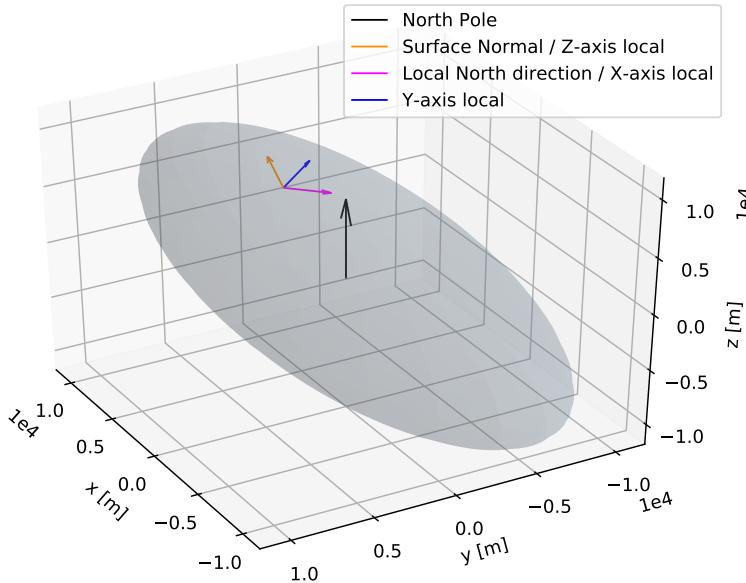
Table B.5: Surface frame verification at the longest edge of the ellipsoid

Unit Vector	Components			Operations
	x	y	z	
Position (\hat{r})	1.0	0.0	0.0	
Normal (\hat{n})	1.0	0.0	0.0	$\hat{r} \times \hat{n} = [0.0, 0.0, 0.0]$
X-axis Surface frame / local North direction (\hat{x})	0.0	0.0	1.0	$\hat{x} \cdot \hat{n} = 0.0$

Now, we perform a more generalized and non-trivial test for the surface frame but this time we choose a more general launch location. A test site was chosen at 30° longitude and 60° latitude. It can be viewed in Figure B.2. Note that at this location, the surface normal vector and the launch position vector will not be aligned. We begin with first verifying that the x-axis basis vector for the surface frame at the current launch site is pointing to North. Another way of stating this is that this vector is tangential to the meridian line, which is running all the way up to the poles. If the x-axis basis vector is indeed tangential to the local meridian, then its x-y plane projection will be collinear to that of the position vector to the launch site.

Table B.6: Surface frame verification for test launch location at 30° Longitude and 60° Latitude, i.e., on the leading edge of the asteroid.

Unit Vector	Components			Operations
	x	y	z	
Position (\hat{r})	0.4330127	0.25	0.8660254	$\hat{x}_x/\hat{r}_x == \hat{x}_y/\hat{r}_y$
X-axis Surface frame / local North direction (\hat{x})	-0.8496329	-0.49053578	0.1936455	$= -1.9621431353$
Normal (\hat{n})	0.05874547	0.27687111	0.95910967	$\hat{n} \cdot \hat{x} = 0.0$

**Figure B.2:** Surface frame depicted for test launch located at 30° Longitude and 60° Latitude. Note that the longitude is measured in anti-clockwise direction from the +X axis in the figure.

We can prove the two projections to be collinear if the x and the y components of the vectors form equal ratios. This is shown in Table B.6. We see that the ratios are equal and hence the x-axis basis vector is pointing to the North direction. Following this, we can also say that the normal vector at this location is formulated correctly since it is perpendicular to the x-axis basis vector (which

in turn is tangential to the meridian line). A vector perpendicular to the meridian line will in-fact be the surface normal vector. Note that, we haven't explicitly verified the y-axis basis vector of the surface frame because it is formed by cross multiplying the normal and the x-axis basis vector. So it inherently remains verified if the latter two are formulated correctly.

Now that we have verified that the surface frame is established correctly, we need to verify that the velocity vector makes the correct declination and azimuth angle with the surface frame. The test launch location is still the same as before and the procedure for verification (explained shortly) is different from how the velocity vector was originally formed. We use the vector dot product definition to compute the angle between the velocity vector and the surface normal vector. This gives us the launch declination angle. We then compute the projection of the velocity vector onto the x-y plane of the surface frame² and then compute the angle between the projection and the x-axis basis vector of the surface frame. The latter is done, again, by using the dot product method. This then gives us the launch azimuth angle. If the computed angles match the ones provided as input for the simulation, then the velocity vector formulation is verified.

Two particle launches were simulated for the aforementioned verification process. The particles were launched with a velocity of 6.0 [m/s] from a point located on the surface of the CDE asteroid at 30° Longitude and 60° Latitude. The first test involved launching particles at declination and azimuth angles of 30° and 45° respectively. The second test involved declination and azimuth angles of 60° and 135° respectively. The results for these test simulations is shown in Tables B.7 and B.8. Figure B.3 shows the orientation of the velocity vector, for the first test, with respect to the launch site surface frame and the ARF.

Table B.7: Launch velocity surface frame angles verification data. Input launch declination = 30° and azimuth = 45°.

Vector	Vector Components			Launch Declination [deg]	Launch Azimuth [Deg]
	x	y	z		
Velocity (\vec{v}) [m/s]	-0.385325	-1.354695	5.832351	30.0	45.0
Unit Normal	0.058745	0.276871	0.959109		
\vec{v} projection [m/s]	-0.690575	-2.793360	0.848671		
Unit x-axis surface frame	-0.8496329	-0.49053578	0.1936455		

Table B.8: Launch velocity surface frame angles verification data. Input launch declination = 60° and azimuth = 135°.

Vector	Vector Components			Launch Declination [deg]	Launch Azimuth [Deg]
	x	y	z		
Velocity (\vec{v}) [m/s]	5.223625	-0.4029416	2.924273	60.0	135.0
Unit Normal	0.058745	0.276871	0.959109		
\vec{v} projection [m/s]	5.047389	-1.2335549	0.046944		
Unit x-axis surface frame	-0.8496329	-0.4905357	0.1936455		

²Given a velocity vector \vec{v} and the surface normal vector \vec{n} , the x-y plane projection of \vec{v} is given as: $\vec{v}_{xy} = \vec{v} - \frac{\vec{v} \cdot \vec{n}}{n^2} \vec{n}$

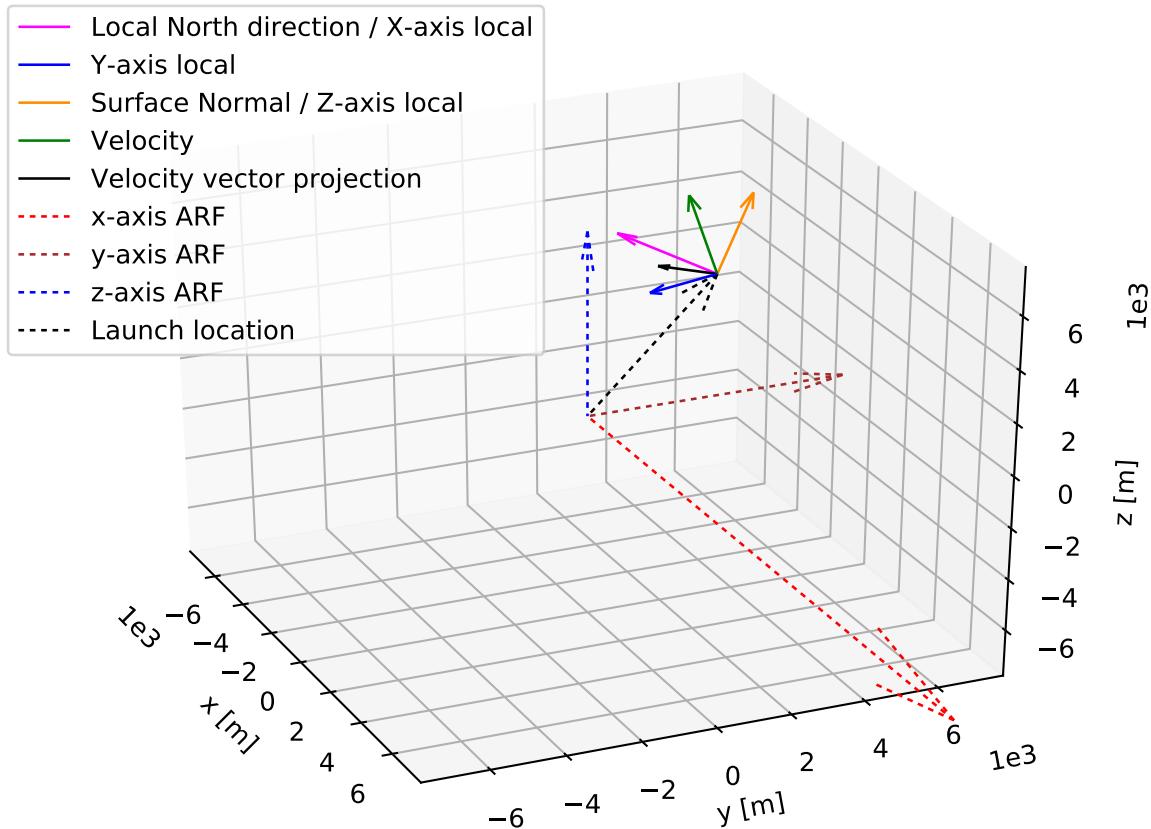


Figure B.3: Schematic representing the surface frame for launch site at 30° Longitude and 60° Latitude along with the velocity vector and the ARF. The diagram gives an intuition on the orientation of the velocity vector. The launch declination and azimuth angles are 30° and 45° respectively.

B.3 REGOLITH ORBITAL MOTION

In this section, we will present results on test simulations that verify and validate the numerical integration of the equations of motion for the regolith in NAOS, and in extension, the numerical integrator, the gravity potential model and the relatively smaller functional aspects of the simulator. The tests were done by using the CDE gravity potential model, which means that gravity perturbations were accounted for, but excluded the Solar perturbations. This is so that we can test the validity of the simulator by observing the conservation properties of the Jacobi Integral and the Keplerian Energy of an orbiting particle. Both of these would not be conserved if perturbations were included in the test.

B.3.1 SPHERICAL ASTEROID

We first test the simulator for a particle launched from the surface of a spherical asteroid of radius 20 km. The launch site was located at 0° Longitude and Latitude. A single particle was launched with a velocity of 10.0 m/s with an azimuth angle of 135° and declination angle of 45°. Note that the CDE potential model was still used for this simulation with all three semi-major axes made equal to the sphere radii. If the gravity potential model is formulated properly, then with all three semi-major axes being equal, the potential model will act like a point-mass model. This would make the gravity field uniform and hence conserve the Keplerian Energy. The Jacobi integral and the Keplerian energy for this test is shown in Figure B.4. From the conservation of these values, we can infer that atleast for a spherical asteroid, the simulator works correctly.

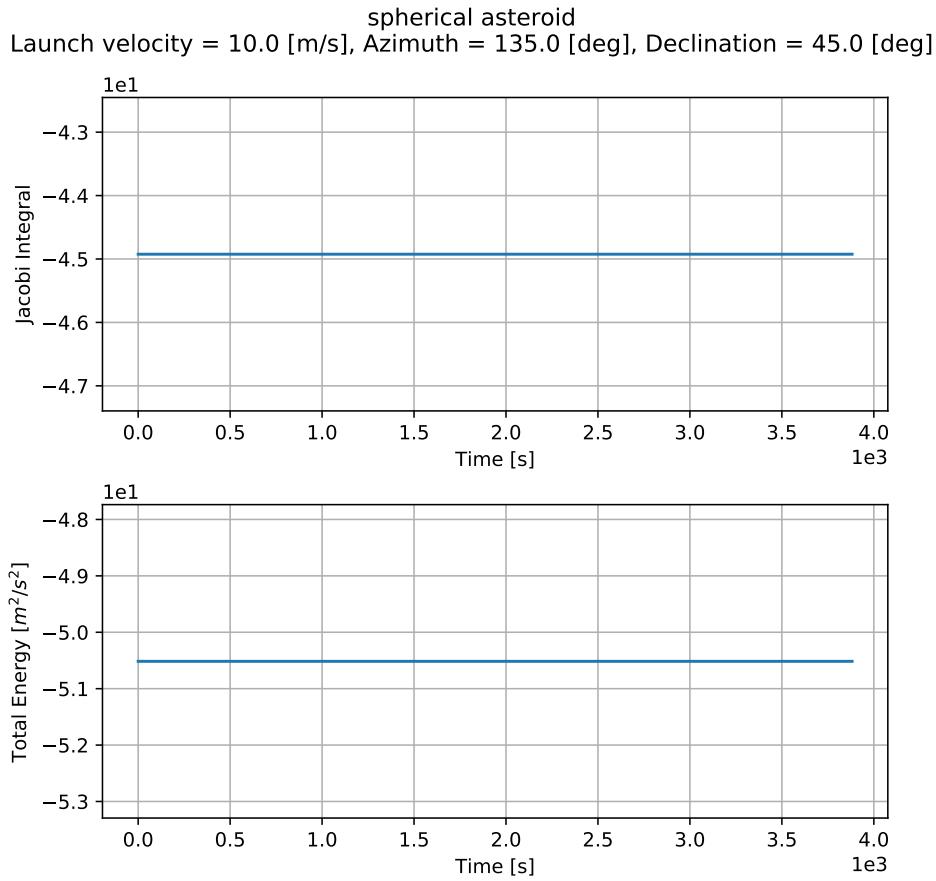


Figure B.4: Jacobi Integral and Keplerian Energy for the regolith launched from the surface of a spherical asteroid at Latitude and Longitude 0° remains conserved.

B.3.2 CDE ASTEROID

We'll perform a similar test as before but this time for a CDE shaped asteroid of semi-major axes $\alpha = 20\text{ km}$, $\beta = 7\text{ km}$, $\gamma = 7\text{ km}$. A single regolith was launched from the surface at site Longitude 30° and Latitude 60°. The particle was launched with a velocity of 10 m/s, azimuth angle of 135° and declination angle of 30°. Now since the gravity potential model is a non-uniform one, unlike that in the case of the spherical asteroid, the Jacobi integral would remain conserved however the Keplerian energy of the particle should not remain conserved. We see this outcome in Figure B.5. This result further validates the simulator since the Jacobi remains conserved, implying that the equations of motion, the gravity potential model, the numerical integrator and several other background functions of the simulator work correctly.

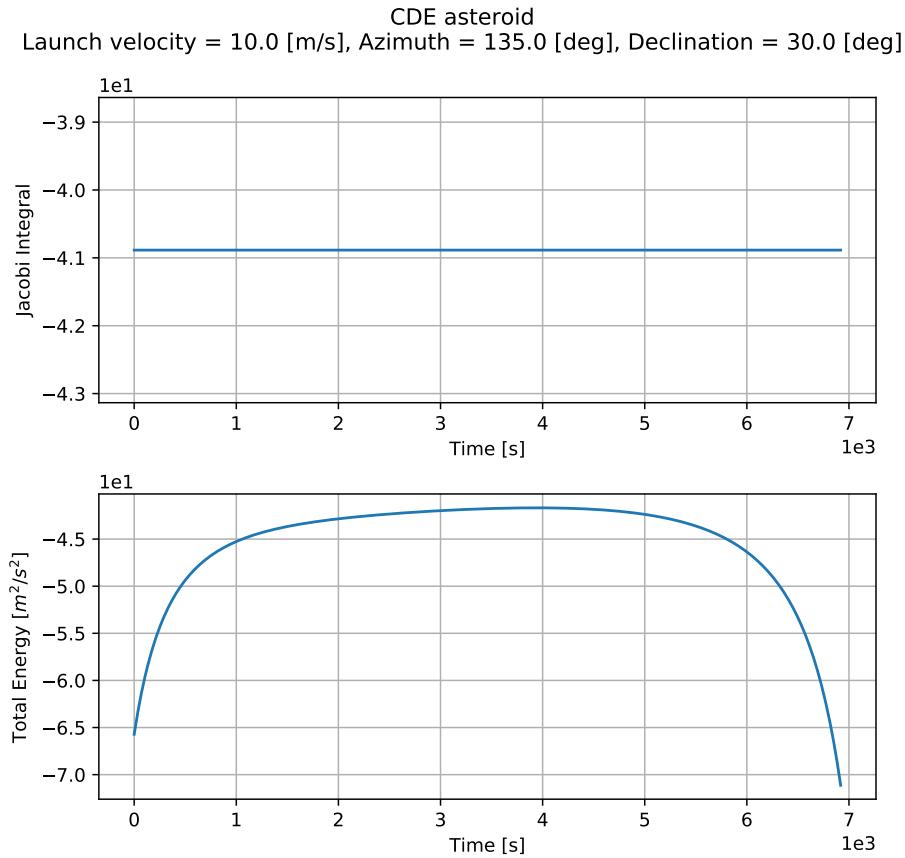


Figure B.5: Jacobi Integral and Keplerian Energy for the regolith launched from the surface of a CDE shaped asteroid at Latitude 60° and Longitude 30°.

Post this, two other tests were performed from the leading edge of the asteroid, one of which resulted in re-impact and the other which resulted in an escape situation. The Jacobi integral for the two simulation was computed which again turned out to be constant throughout the duration of the respective trajectories. The results and initial conditions for the re-impact case are shown in Figure B.6 and for the escape case are shown in Figure B.7. The launch declination in both cases is 45°. The results further validate the functionality of the simulator.

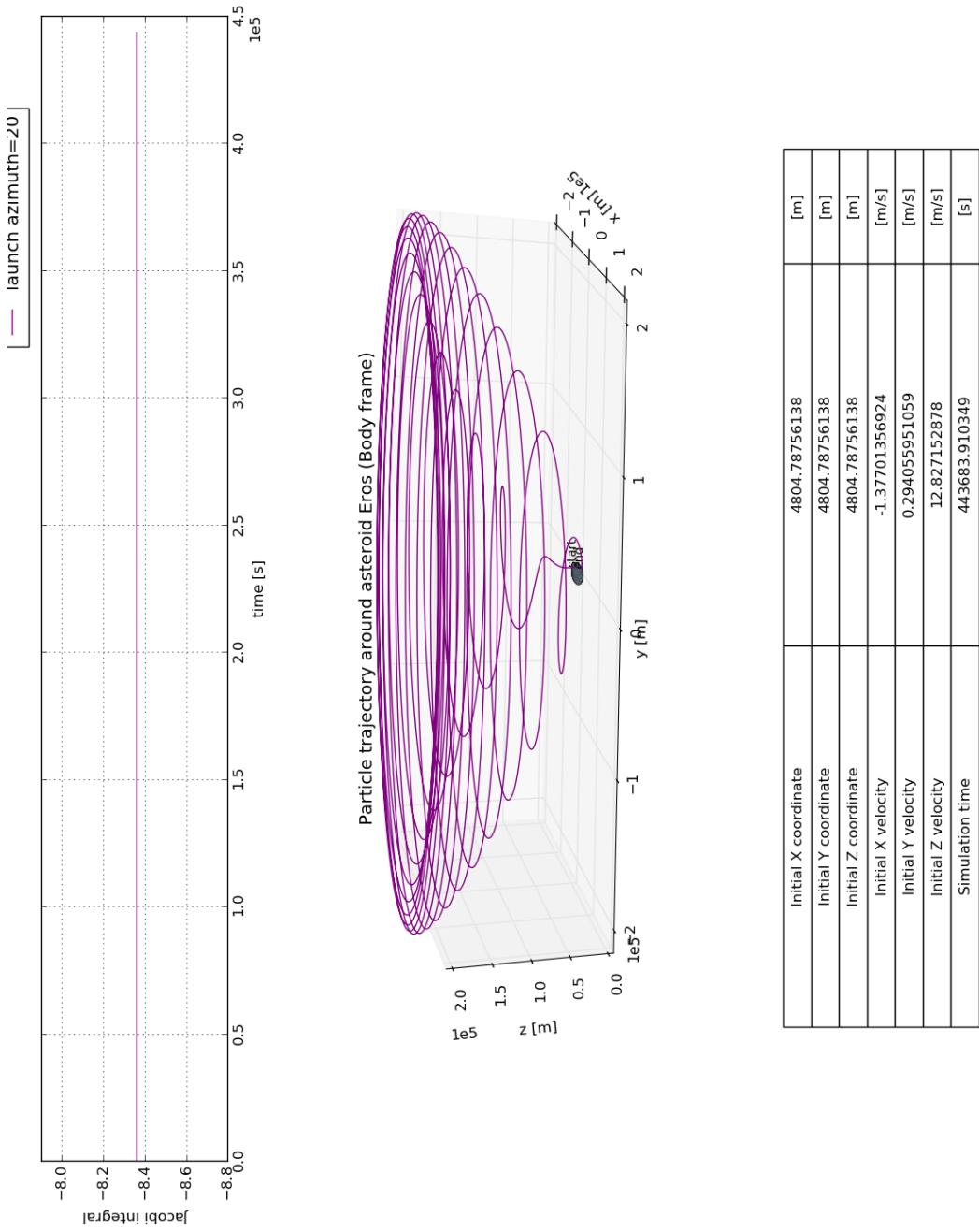
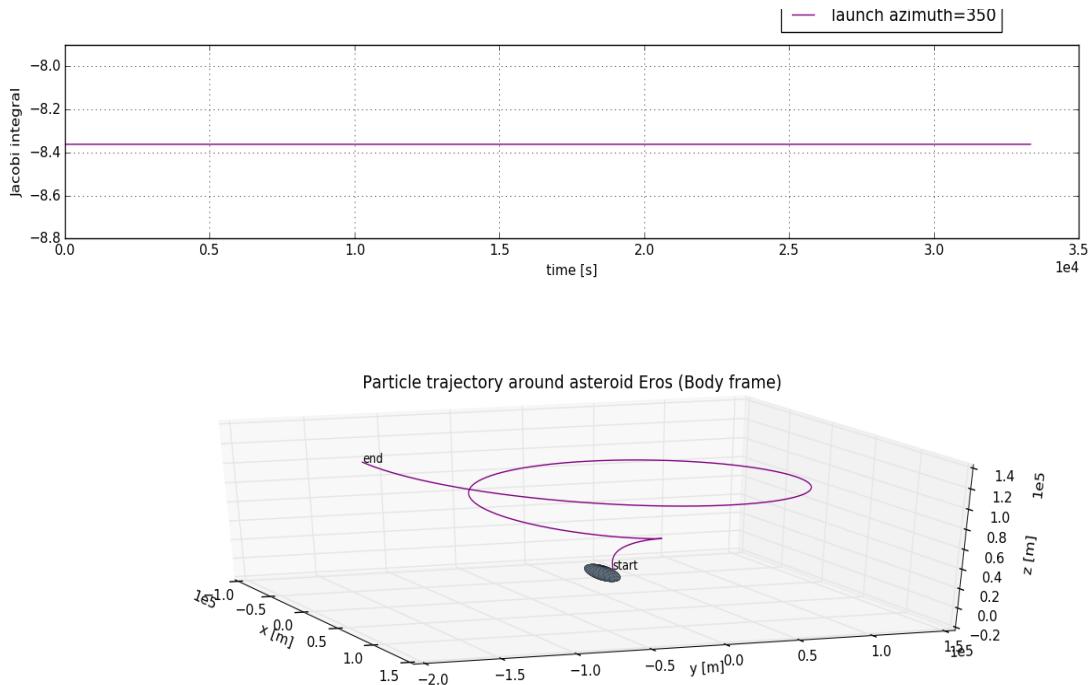


Figure B.6: Jacobi Integral for the regolith launched from the leading edge of a CDE shaped asteroid which eventually re-impacts the surface of the asteroid.



Initial X coordinate	4804.78756138	[m]
Initial Y coordinate	4804.78756138	[m]
Initial Z coordinate	4804.78756138	[m]
Initial X velocity	-5.50162085006	[m/s]
Initial Y velocity	2.15435970563	[m/s]
Initial Z velocity	11.4721135153	[m/s]
Simulation time	33330.0	[s]

Figure B.7: Jacobi Integral for the regolith launched from the leading edge of a CDE shaped asteroid which eventually escapes the gravitational attraction of the asteroid.

B.3.3 INTEGRATOR PERFORMANCE

The integrator used in our simulator NAOS was from an external library called *boost*. We use this library because coding higher order numerical integrators is an extremely daunting task which would have swayed away time and focus from the thesis at hand. On top of that, the boost library provides verified integrator subroutines (among other things) and hence this reduces our task to just verify that it has been used properly in our simulator. From the results in the previous two subsections, we can say that the integrator has been properly accommodated in NAOS otherwise the behavior of the Jacobi integral and the Keplerian energy would have been different. Here, we will briefly discuss the performance of the integrator under different configurations of itself, for the same particle launch conditions as mentioned in Section B.3.2. But first we'll give a brief description of the integrators adaptive behavior before we can look at some numbers on the performance.

The integrator used is a Runge-Kutta-Fehlberg78, which is an 8th order integrator with the 7th order used for error control. The integrator uses an adaptive step size, which means that the step size for integration, from one epoch to another, changes continuously depending on the magnitude of in-

tegration error at each step. What this means is that the number of steps taken to integrate from an initial epoch to the final one changes, for every instance of integration in a sequence. So to propagate the state vector from, say, time t_0 to t_1 , the integrator first performs a single step of integration based on an initial guess for the step-size. This is done first using an order 8 process then again using a 7th order process. The difference in the outcome between the two is then treated as the numerical error for that step. This process of error estimation of-course repeats for every other step of integration performed, to go from time t_0 to t_1 . The estimated error is compared with the following equation:

$$abs_{tol} + rel_{tol} \times (X + (dt * dXdt)) \quad (\text{B.1})$$

where abs_{tol} and rel_{tol} are the absolute and relative error tolerances of the integrator, and their values can be set by the user; X is the state vector, dt is the time step for integration, and $dXdt$ is the first order differential equation for the state vector (so basically the equations of motion in our case). Now if the estimated error is smaller than Equation (B.1), the integration step is accepted and the integrator moves on to the next step. If, however, the estimated error is larger, then the step size is reduced and the integration is performed again. There are other processes that run internally within the subroutine, that ensure that the step-size does not become too small or too large, the details of which can be found at ([Boost, "Integrator"](#)).

Table B.9 gives values for certain metrics that showcase the variation in performance of the Runge-Kutta-Fehlberg78 integrator. Note that we propagated the trajectory of the regolith for the same initial conditions as mentioned in Section B.3.2. The metric CPU time refers to the total time taken by the computer processor to integrate the entire problem from start to end. The total CPU time shouldn't be taken at face value but rather at the order of magnitude because background processes in the computer could delay the time taken to perform a given simulation. The entire simulation, from the start till the end, is broken down into a series of smaller integration instances. For each instance then, the number of steps to perform the integration changes since the step-size keeps on changing. Thus, the column *max. step* refers to the maximum number of steps undertaken for any integration instance in the entirety of the simulation. Similarly, the column *min. step* gives the values for the least number of steps. Each row in Table B.9 corresponds to one entire simulation performed with the given absolute and relative tolerances.

Table B.9: Variation in integrator performance for different error tolerance values.

Absolute Tolerance	Relative Tolerance	CPU time [s]	Max. Steps	Min. Steps
10^{-2}	10^{-2}	0.54	6	3
10^{-6}	10^{-6}	0.47	6	3
10^{-15}	10^{-15}	0.41	6	3
10^{-20}	10^{-20}	2.43	2468	5

It is important to note that the integrity of the simulation from the dynamics point of view was not hampered for any of the combinations of absolute and relative tolerances given in Table B.9. They all gave the same result as that in Figure B.5. A logical inference to be drawn from this is that even when the tolerance is relatively large, the simulation results turn out to be the same because the estimated error in integration itself is very small in the first place. We see a visible difference in the performance only when the tolerances are made extremely small, such as 10^{-20} , which ultimately causes the integrator to perform computations at much smaller step-sizes because now the estimated error gets larger in comparison to Equation (B.1). Not shown in Table B.9, but the tolerances were further reduced to 10^{-30} at which point the simulation got extremely slow and never

ceased within a reasonable amount of time. Thus extremely small tolerances, i.e. beyond 10^{-15} , should not be used for the purposes of this thesis since we are simulating several thousand particles at the same time. We ultimately decided to use an absolute and relative tolerance of 10^{-15} since it gave the same performance as any other higher tolerance value.

B.4 SOLAR PERTURBATIONS

In this section we will provide results on tests performed to validate the perturbing force models. The test data was taken from, the already verified, unit test files of Technical University Delft Astrodynamics Toolbox (TUDAT) ³. We benchmark the force models used in NAOS by performing tests with data from TUDAT and data obtained from some simplified hand-based calculations as well. The subroutines for the force models in NAOS were modular enough and no changes were made to the function codes to accommodate the validation process.

B.4.1 SOLAR THIRD-BODY EFFECT

We'll begin by presenting validation data for the STBE perturbing acceleration. The position vector of the target location where the perturbing acceleration had to be calculated, and the position vector to a random perturbing body, are both mentioned with respect to some common arbitrary frame of reference. The definition of the latter does not matter or affect the computation within the STBE force model. The gravitational parameter of the perturbing body is $4900.0 \times 10^9 \text{ m}^3/\text{s}^2$. The computed acceleration values matched those provided in the TUDAT unit test files, shown in Table B.10, thus validating the STBE force model for a more generalized 3D data.

Table B.10: Validation data for testing the STBE force model, taken from unit test files in TUDAT. The gravitational parameter of the perturbing body is $4900.0 \times 10^9 \text{ m}^3/\text{s}^2$.

Vector	Component		
	x	y	z
Target position [m]	-40000000.0	9000000.0	-9500000.0
Perturber position [m]	25000000.0	-380000000.0	-55000000.0
Perturbing acceleration [m/s^2]	2.93946e-06	2.22539e-06	1.16801e-06

The second test was a more simplified one and uses hand-based calculations to verify the software routine. This was done to test if the routine performs correctly even for an edge case. The test considers a planar situation wherein the regolith is on the positive x-axis with respect to the asteroid (consider looking at Figure 4.5 to visualize the set up) and the Sun is in the equatorial plane of the asteroid on the negative x-axis. The position vectors for the two bodies and the corresponding acceleration values, both hand-calculated and software computed, are shown in Table B.11.

Table B.11: Validating the STBE force model using hand-calculated acceleration values for a specific edge case.

Vector	Component		
	x	y	z
Target position [m]	25000.0	0.0	0.0
Perturber position [m]	-1.0 AU	0.0	0.0
Perturbing acceleration [m/s^2]	Hand-calculated	1.98201e-09	0.0
	Software computed	1.98201e-09	-3.64089e-25

There is an extremely small round off error in the y-component of the software computed perturbation acceleration but apart from that the software values match the hand-calculated ones in

³TUDAT is an open source astrodynamics toolbox, developed and maintained by the department of astrodynamics and space missions at the Delft University of Technology and the toolbox can be found at <https://github.com/tudat/>.

terms of both magnitude and direction.

B.4.2 SOLAR RADIATION PRESSURE

We'll now present validation data for the SRP force model. The first test assumes a spacecraft near Venus. The parametric data for the test was again taken from the TUDAT unit test files and is shown in Table B.12. The acceleration due to SRP computed from the software routine matched in direction and magnitude with the test data. The position vector goes from the target to the Sun and is defined with respect to an arbitrary frame. The definition of the frame does not matter for the software routine to calculate the perturbing accelerations, apart from the fact that the values will be defined with respect to the arbitrary frame.

Table B.12: Validation of SRP model using test data from TUDAT. The test assumes a satellite somewhere near Venus and considers a planar case.

Parameter	Value
Target to Sun position vector [m] (x, y, z)	(77432181578.46405, 77432181578.46405, 0.0)
Target emissivity	0.5
Solar incident area [m ²]	0.005
Target mass [kg]	0.0022
Solar constant	1.0205062450596109e+17
Perturbing acceleration [m/s ²] (x, y, z)	(-2.05148e-05, -2.05148e-05, 0.0)

The second test, again taken from TUDAT, assumes a random location for the target in 3D, thus providing a more generalized test scenario. The incident area and target mass are also extremely exaggerated. The parametric data used for the test and the output acceleration values are shown in Table B.13.

Table B.13: Validation of SRP model using test data from TUDAT. The test assumes an exaggerated target body at a random location and considers a general 3D scenario.

Parameter	Value
Target to Sun position vector [m] (x, y, z)	(94359740.25, 90831886.1, 14668782.92)
Target emissivity	0.4058
Solar incident area [m ²]	514701.9505
Target mass [kg]	1.0
Solar constant	1.0205062450596109e+17
Perturbing acceleration [m/s ²] (x, y, z)	(-3.04373e+06, -2.92993e+06, -473166)

The perturbing accelerations computed from the software routine for SRP in NAOS matches the test data from TUDAT, thus validating the software model. Just like for STBE, we present a case of validation against hand-calculated data for a set of very simple parametric values. The results and test data are given in Table B.14. There is an extremely small round-off error in the x-component of the computed acceleration, but the magnitude and direction matches that of the hand-computed values. Thus with this final test, we can say that the software is verified.

Table B.14: Validation of SRP model in NAOS for an edge case against hand-calculated data.

Parameter	Value				
Target to Sun position vector [m] (x, y, z)	(0.0, 0.1 AU, 0.0)				
Target emissivity	1.0				
Solar incident area [m^2]	1.0				
Target mass [kg]	1.0				
Solar constant	1.0e+17				
Perturbing acceleration [m/s^2] (x, y, z)	<table border="1"> <tr> <td>Hand-calculated</td> <td>(0.0, -0.000893674, 0.0)</td> </tr> <tr> <td>Software computed</td> <td>(-5.47218e-19, -0.000893674, 0.0)</td> </tr> </table>	Hand-calculated	(0.0, -0.000893674, 0.0)	Software computed	(-5.47218e-19, -0.000893674, 0.0)
Hand-calculated	(0.0, -0.000893674, 0.0)				
Software computed	(-5.47218e-19, -0.000893674, 0.0)				

B.5 REGOLITH FINAL FATE

Now that we have verified the gravity model, the orbital dynamics and the perturbing force models, the last major validation that we need to perform is to check whether the final fate of a regolith is correctly determined. We do the check for test simulations run from the leading edge of the asteroid while including perturbations. The launch location was at Longitude 30° and Latitude 60°. Multiple regolith were launched from the site with all possible combinations of the following launch conditions: velocities ranging from 10 m/s to 13 m/s, launch azimuth angles varying in steps of 45°, and launch declination angles of 30° and 60°. The initial Solar phase angle was 315°.

This set of launch site and initial conditions provides a very generalized scenario for testing the eventual outcome of the regolith. Figure B.8 shows the results for all recorded escape cases from the test simulation. Every particle that has escaped satisfies the necessary condition of having eccentricity greater than or equal to one and energy being positive. Note that these values are given for the final epoch in the simulation i.e. when any of the possible final fates is realized and the simulation is stopped.

We can validate the re-impact situations by evaluating the triaxial ellipsoid equation for the state vector at the final epoch of the simulation. This was done for all cases that had been identified, by the simulator, to have re-impacted the surface of the asteroid. If the solution turns out to be zero, then we know that the particle is on the ellipsoid itself. In Figure B.9, we do see that all impacted particles provide the desired solution to the triaxial ellipsoid equation. The results are shown only for a subset of re-impact cases, i.e., only for those regoliths whose initial velocity was 13 m/s.

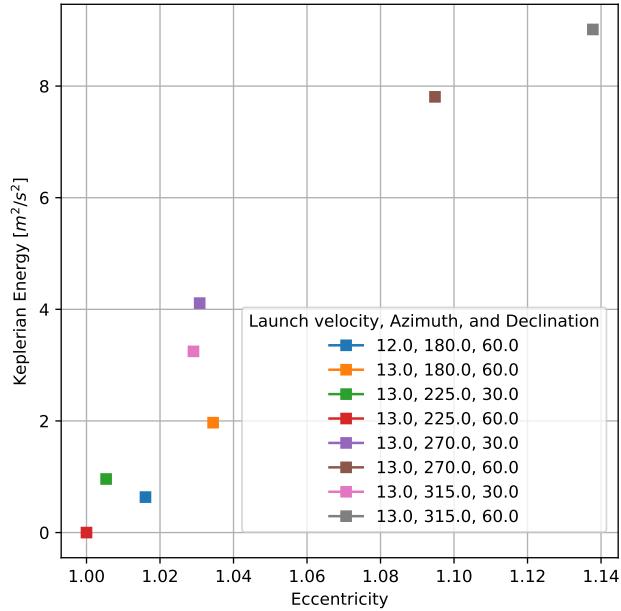
CDE asteroid, Launch Latitude = 60° , Launch Longitude = 30° 

Figure B.8: Energy v/s eccentricity plot for regolith that eventually escaped after being launched from the surface of the asteroid. When the energy is positive and eccentricity is greater than or equal to one, the regolith escapes. In the legend, the launch velocity and the two angles are expressed in m/s and degrees respectively.

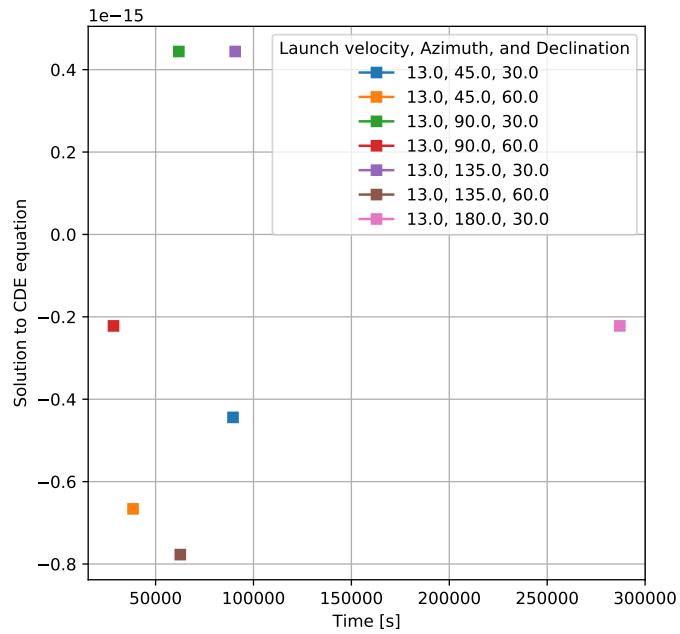
CDE asteroid, Launch Latitude = 60° , Launch Longitude = 30° 

Figure B.9: Solution to the triaxial ellipsoid equation for a subset of re-impacted particles. This solution is obtained for the state vector at the final epoch of the simulation, for the given subset of re-impacted particles. In the legend, the launch velocity and the two angles are expressed in m/s and degrees respectively.

In the test just performed, from the leading edge of the asteroid, we did not obtain any capture cases. To validate this special final fate, we ran another simulation from the longest edge of the asteroid. Note that a capture case is indirectly detected when the simulator propagates the regolith and it doesn't escape or re-impacts the surface of the asteroid. So we perform an internal passive test for a known capture case, wherein we check that the particle doesn't have escape characteristics when far away from the asteroid and neither does it have re-impact characteristics when extremely close to the asteroid.

A single regolith was launched from Longitude and Latitude 0° , with launch velocity magnitude of 10 m/s, azimuth angle of 45° and declination angle of 45° . The initial Solar phase angle was 315° . Figure B.10 shows the osculating Keplerian energy and eccentricity of the regolith when it is far away from the asteroid, i.e., when the range to the particle is equal to or beyond 10 times the largest semi-major axis of the asteroid, which is 200 km in this particular case. When far away, the regolith always maintains a negative total Keplerian energy and an eccentricity less than 1.0, thereby staying bounded to the asteroid.

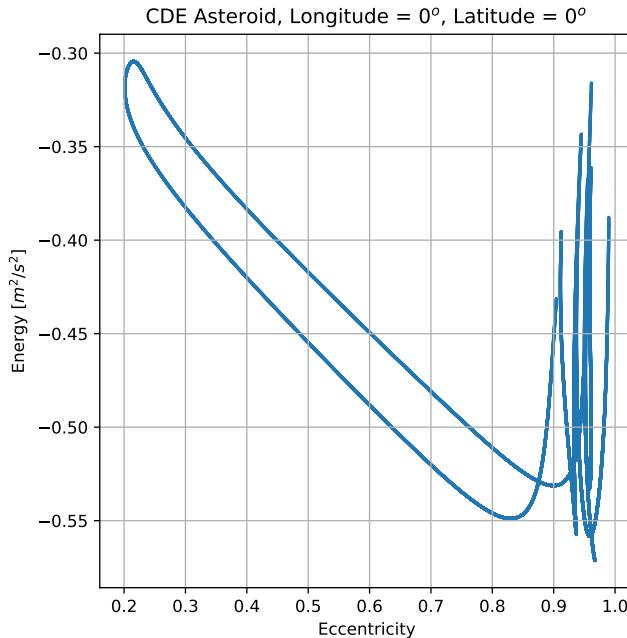
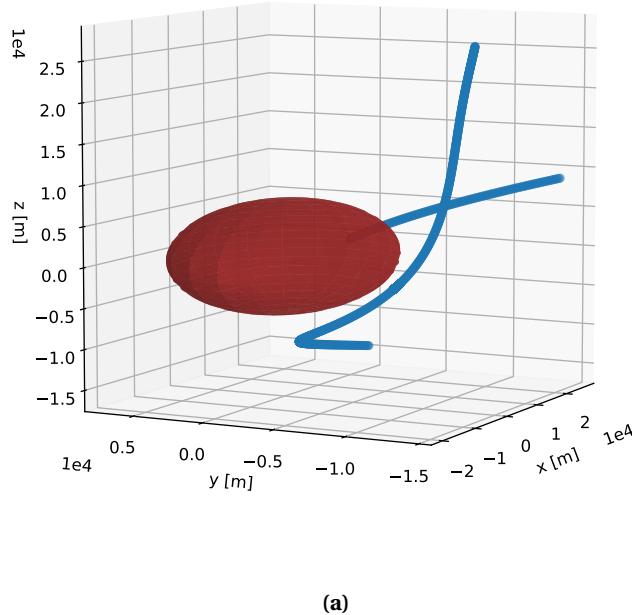
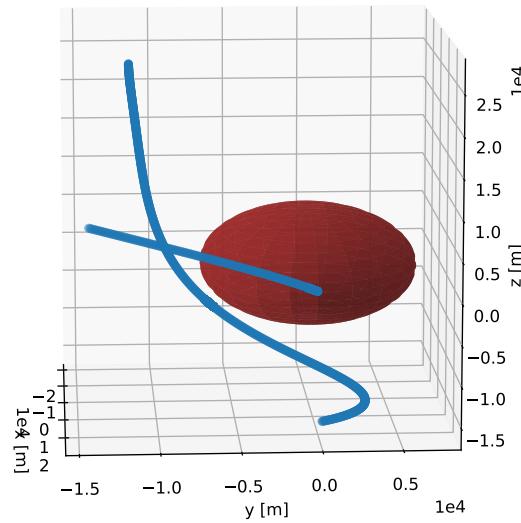


Figure B.10: Osculating Keplerian energy versus eccentricity plot for the regolith with temporary capture as its final fate. The data points plotted here are for the case when the particle has a range beyond 10 times the largest semi-major axis of the CDE (i.e. 200 km in this test). This check was performed to ensure that the particle does not have escape characteristics when far away from the asteroid.

The second indirect test is to check if the particle ever interacts with the surface of the asteroid, that goes unnoticed by the simulator for any reason at all. In this regard this test case is special since we witness that the regolith comes extremely close to the asteroid. We do this by plotting the trajectory data points of the regolith, expressed in ARF, when it lies within a range of 1.5α (i.e. ≤ 30 km). Figure B.11 shows two different views of a part of the 3D trajectory of the captured regolith. It's easy to see that although the particle comes extremely close to the asteroid, it does not collide with it. Note that in Figure B.11b, the trajectory points that appear to connect to the longest edge of the asteroid actually depict the launch of the particle and not a re-impact scenario.



(a)



(b)

Figure B.11: 3D trajectory data points for a capture scenario. The data points plotted here are for the case when the particle has a range $\leq 1.5\alpha$, i.e., 1.5 times the largest semi major axis of the asteroid. This check was performed to ensure that the particle has not experienced a re-impact situation that went unnoticed by the simulator.

B.6 CONCLUSION

In this chapter, we presented results for verification and validation of the simulator NAOS. The tests, and the results from it, proved the authenticity of NAOS as we verified its high-level functioning layers such as the perturbing force models, gravity field model, numerical propagator, regolith orbital dynamics, regolith launch parameters and final fate. Validating the aforementioned, in general, validates the functioning of the simulator developed for thesis. However, it must be noted that the high-level validation tests may, often, not act as proofs of correct performance for the lower functional layers of any simulator and hence individual unit tests and sanity checks should be written within the simulator itself to check for bugs that may appear under special circumstances.

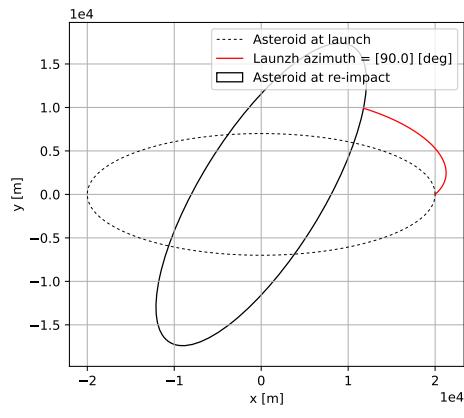
In this regard, it should be noted that NAOS has certain unit test files to validate the lower functional layers using external verified data, for example the cubic root solver for the gravity potential model, the Cartesian to Kepler converter etc. The unit test file is an independent piece of code that checks the performance of, say, the cubic root solver against known data. This unit test file then not only validates its parent function, but also ensures that in future if the parent function is ever modified, then the changes also conform to a correct output value when matched against the same external verified data as before.

In addition to this, the code for NAOS was developed sequentially and by using modular functions and C++ template files that enable unit testing of several features of the code without changing its parameters manually to accommodate external data. Thus the parent functions are never (manually) hampered to verify its core functionality, otherwise there is always a risk of making an error when changing the code back to its original form. There are several internal sanity checks as well at various points within the NAOS code to ensure bugs can be spotted before compromising the integrity of the simulated data. Some of these sanity checks are re-run in the Python codes of NAOS that perform data analysis.

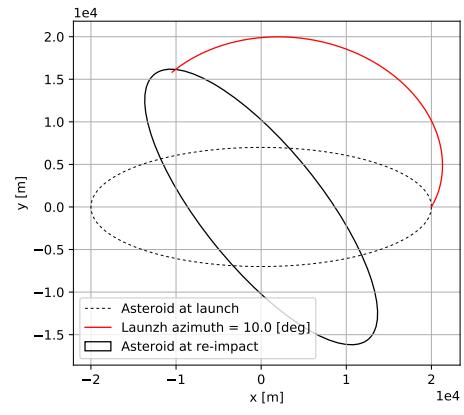
C

EXTRA FIGURES

This appendix contains figures which are used to support the explanation of certain results, arguments and conclusions in the main part of the Thesis report.

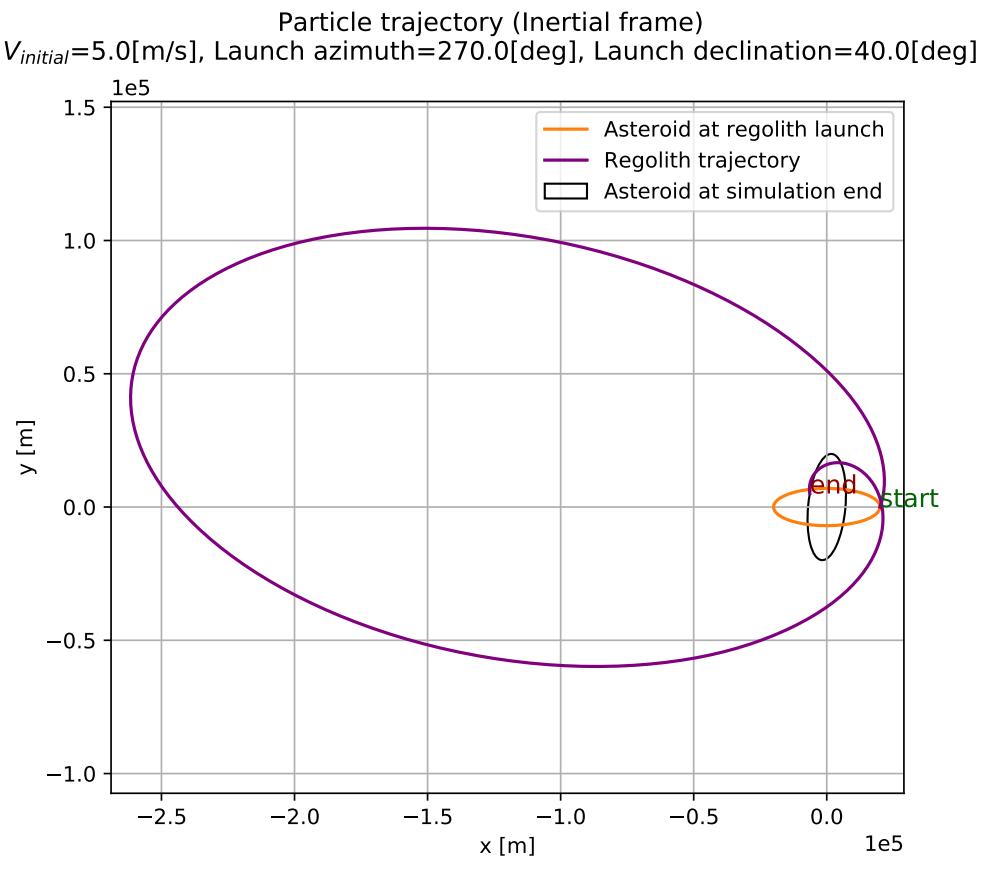


(a)

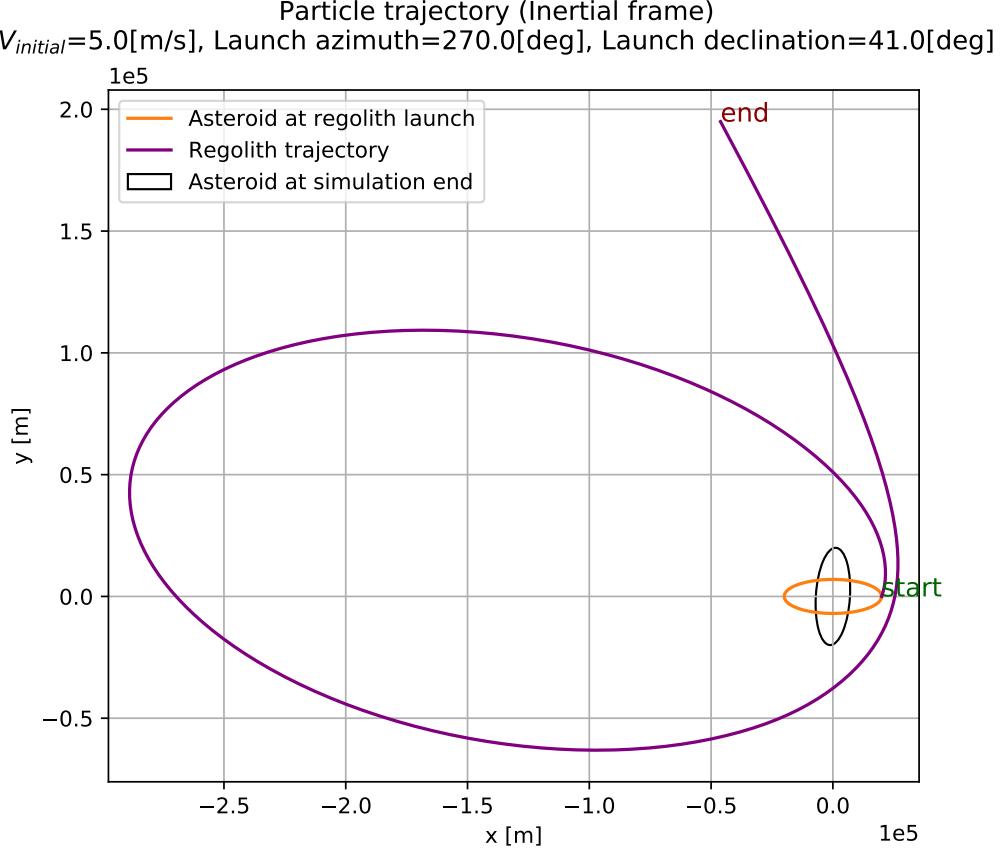


(b)

Figure C.1: Re-impact trajectory for regolith launched from the longest edge of the asteroid at 5 m/s. Trajectories for two different launch azimuths is shown, i.e., 90° and 10° .



(a)



(b)

Figure C.2: Trajectory plots for regolith launched from the longest edge of the asteroid at 5 m/s, at two different launch declinations (a) 40° and (b) 41° are shown.

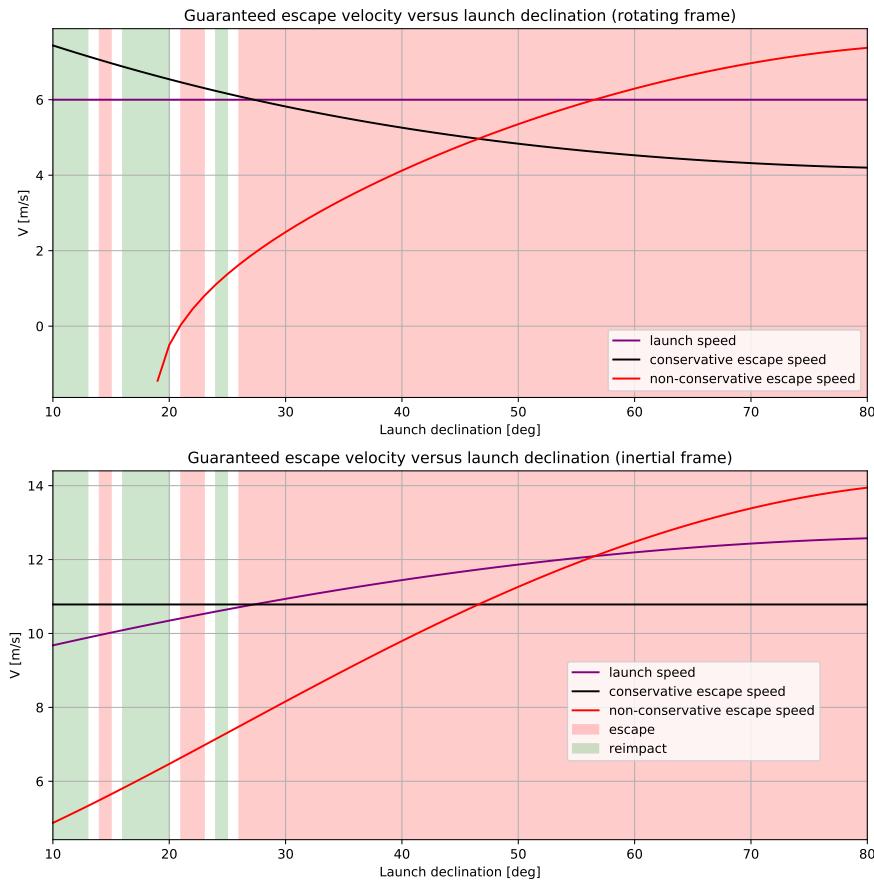


Figure C.3: A non-applicable NCGES curve for $q_{\infty} = 0.7$. The curve in the rotating frame or the ARF shows negative speed values which is not valid.

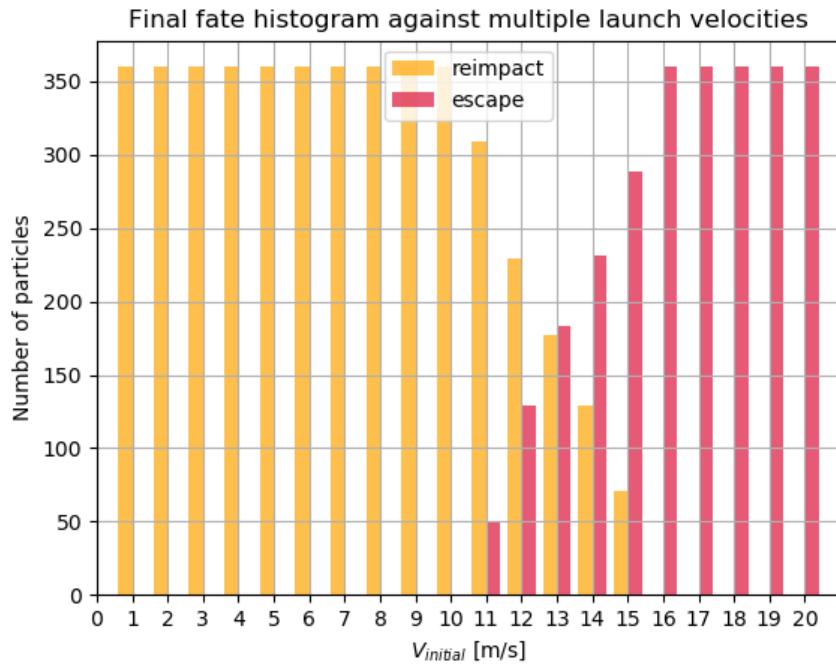


Figure C.4: Re-impact/escape histogram for particles launched from the equator of a spherically shaped asteroid of radius 10 km. No Solar perturbations were considered in the simulation.

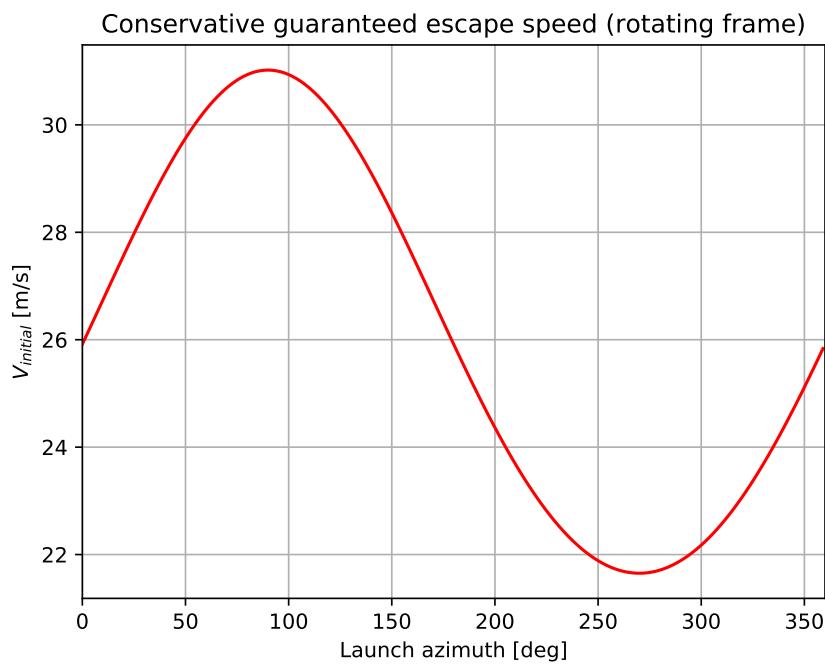


Figure C.5: CGES variation with launch azimuth for particles launched from an equatorial site on a spherical asteroid of radius 20 km.

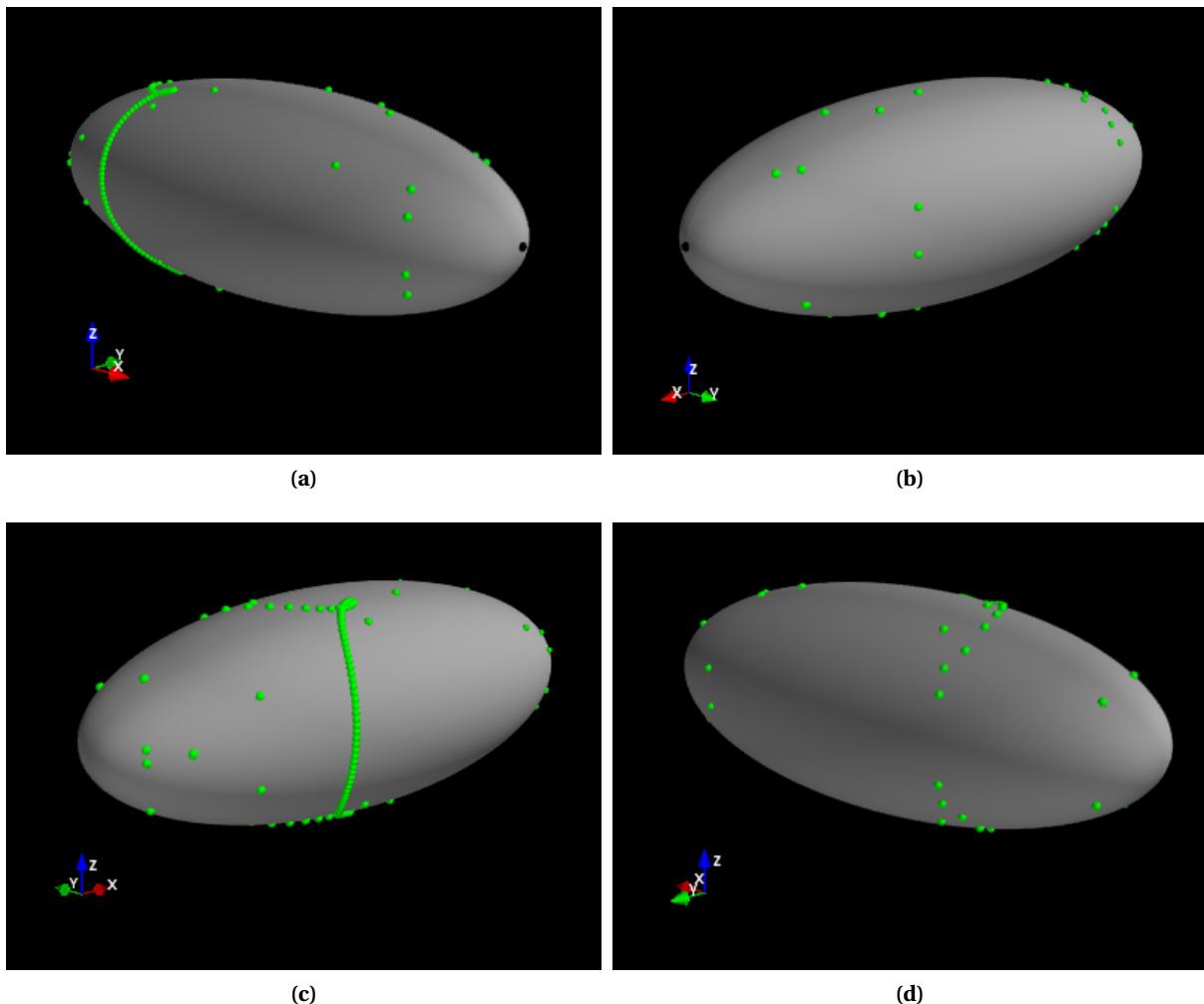


Figure C.6: 3D view, from four different angles, of regolith re-impact locations for particles launched with a velocity of 9 m/s. The green points represent the re-impact sites and the singular black dot represents the launch site.

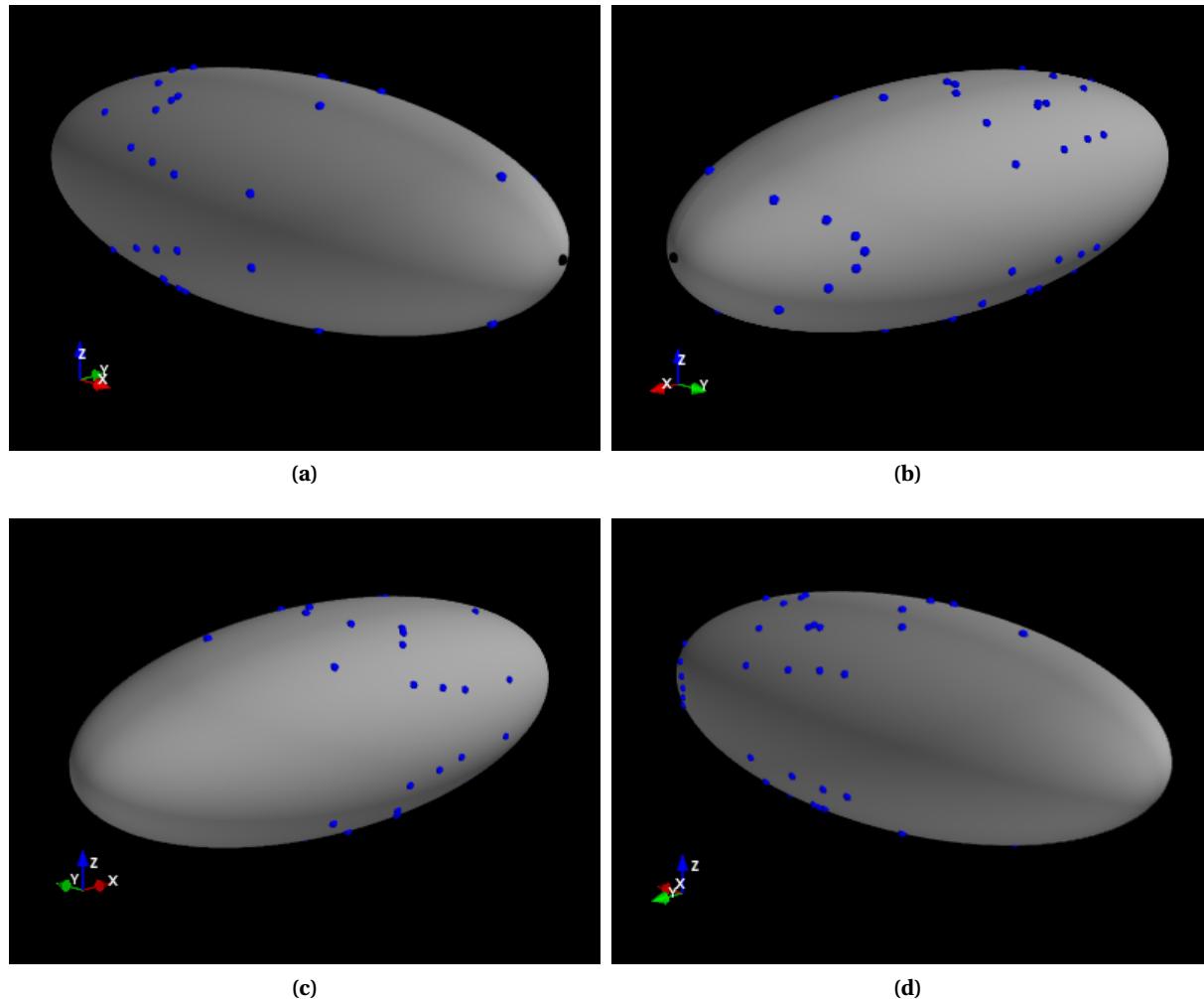


Figure C.7: 3D view, from four different angles, of regolith re-impact locations for particles launched with a velocity of 13 m/s. The blue points represent the re-impact sites and the singular black dot represents the launch site.

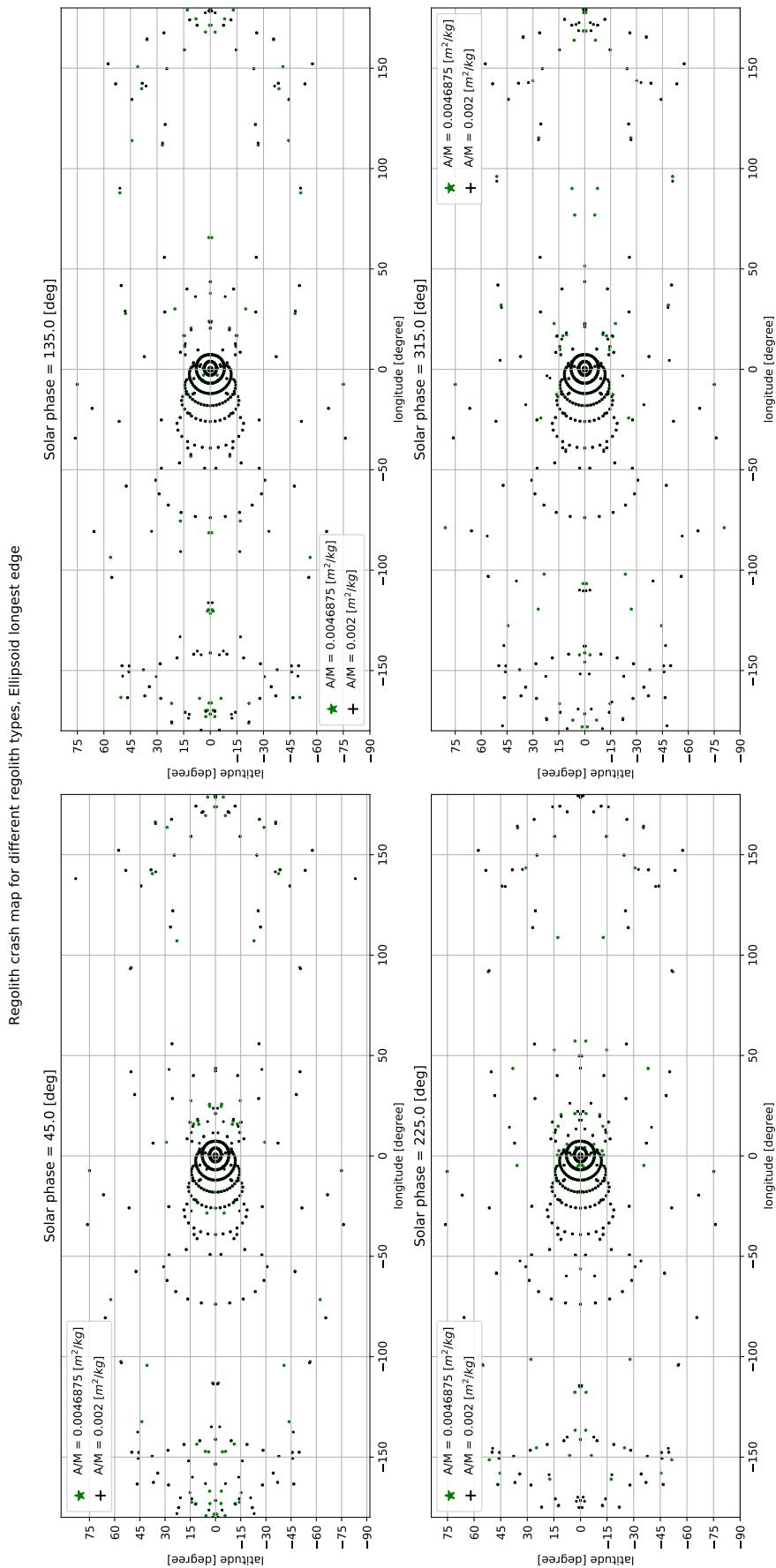


Figure C.8: Re-impact locations for regoliths of different density but same size (i.e. 5 cm). The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

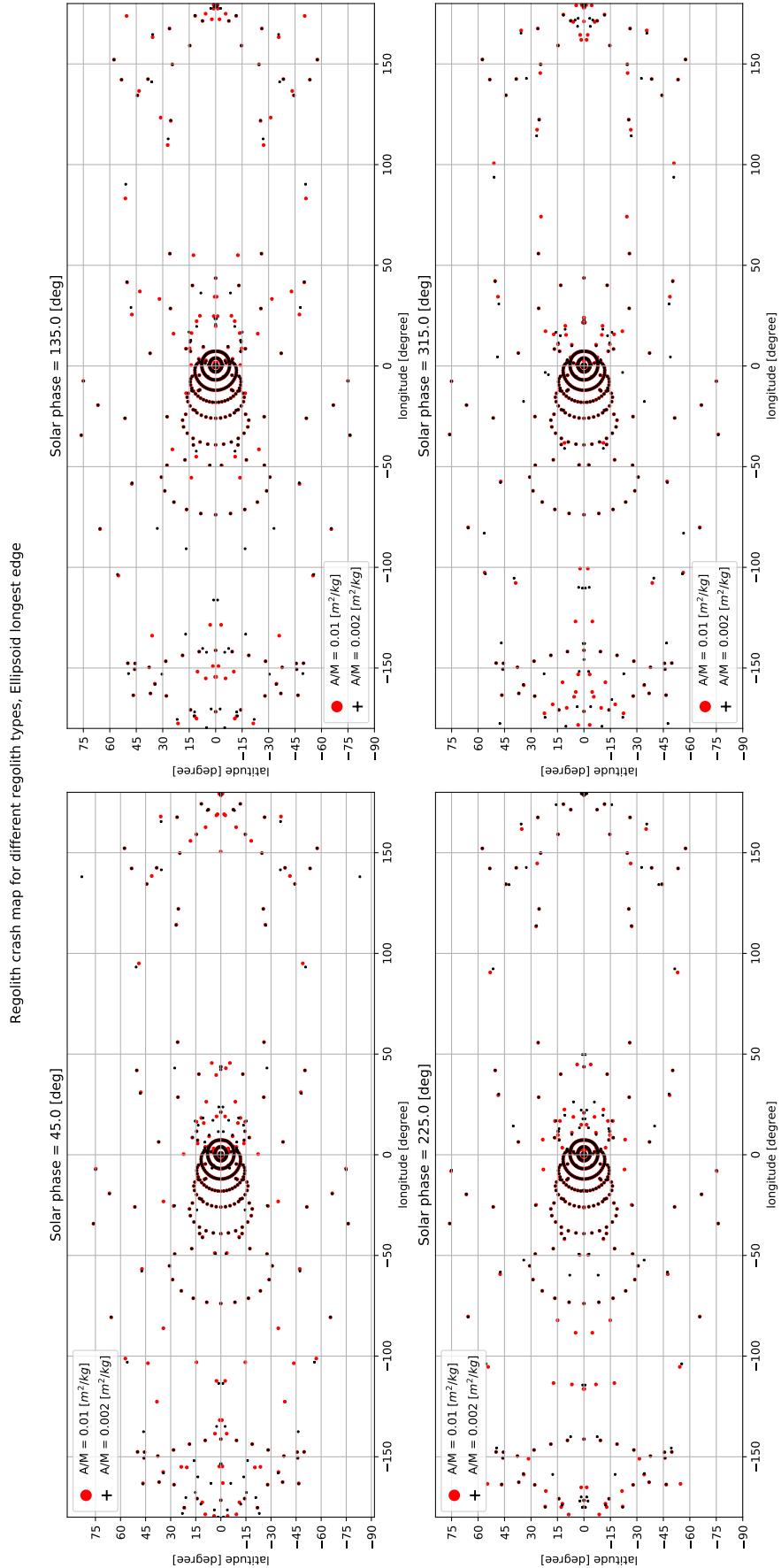


Figure C.9: Re-impact locations for regoliths of different sizes but same density (i.e. 7.5 g/cm^3). The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

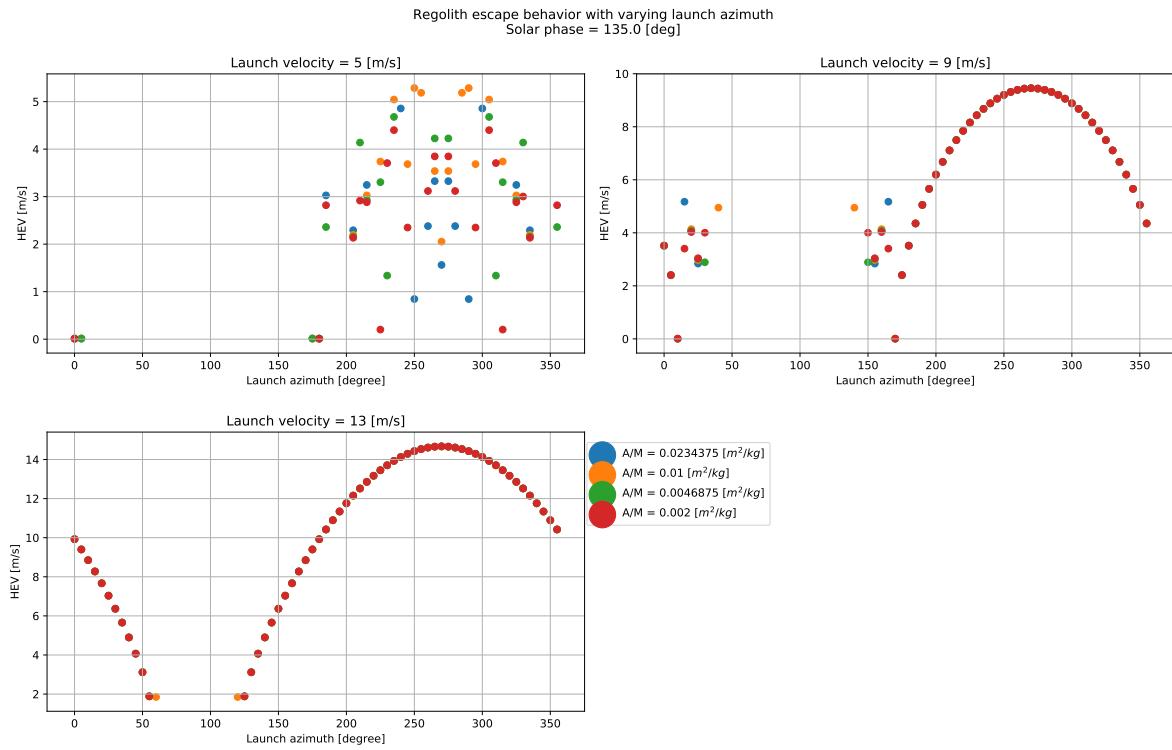


Figure C.10: HEV, shown for all regolith types used in this thesis, but only for three specific launch velocities. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

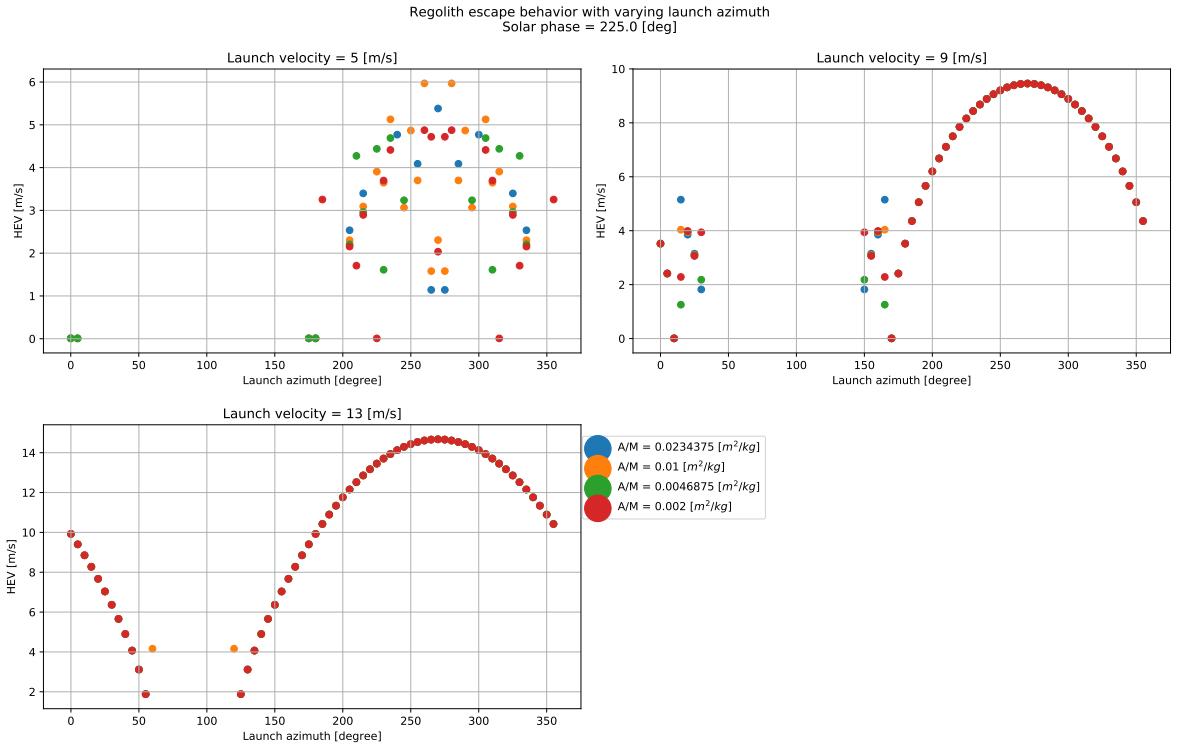


Figure C.11: HEV, shown for all regolith types used in this thesis, but only for three specific launch velocities. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

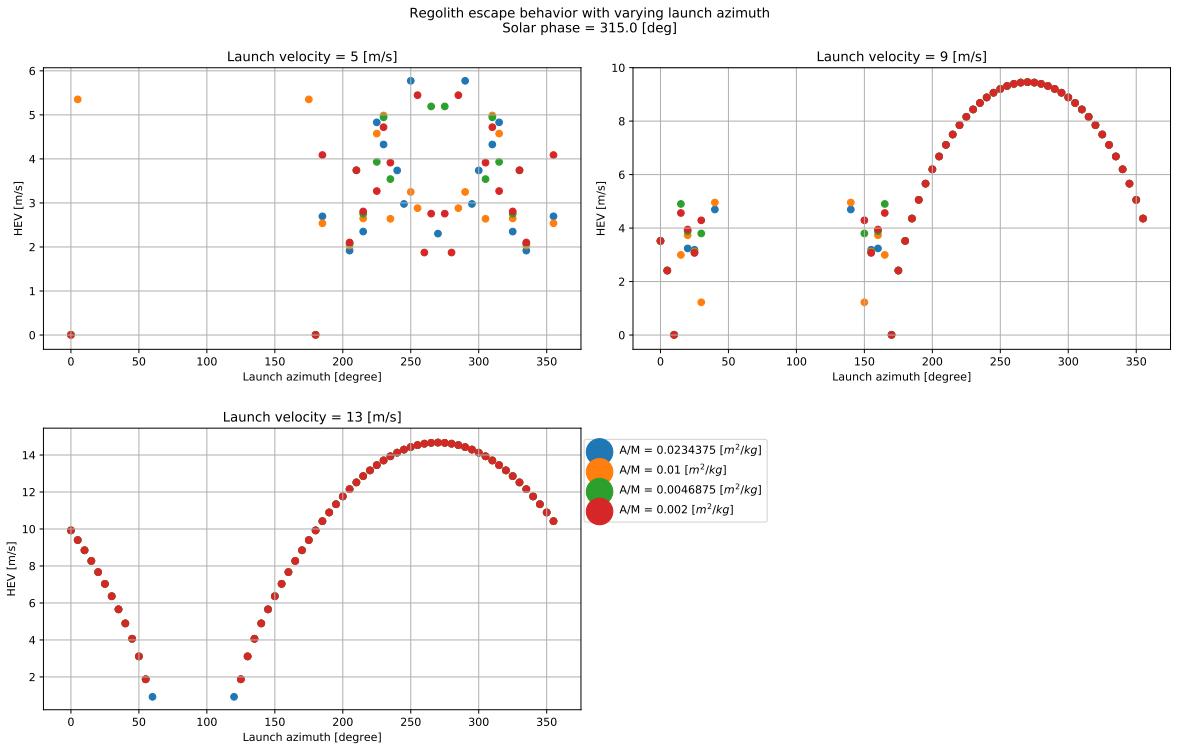


Figure C.12: HEV, shown for all regolith types used in this thesis, but only for three specific launch velocities. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

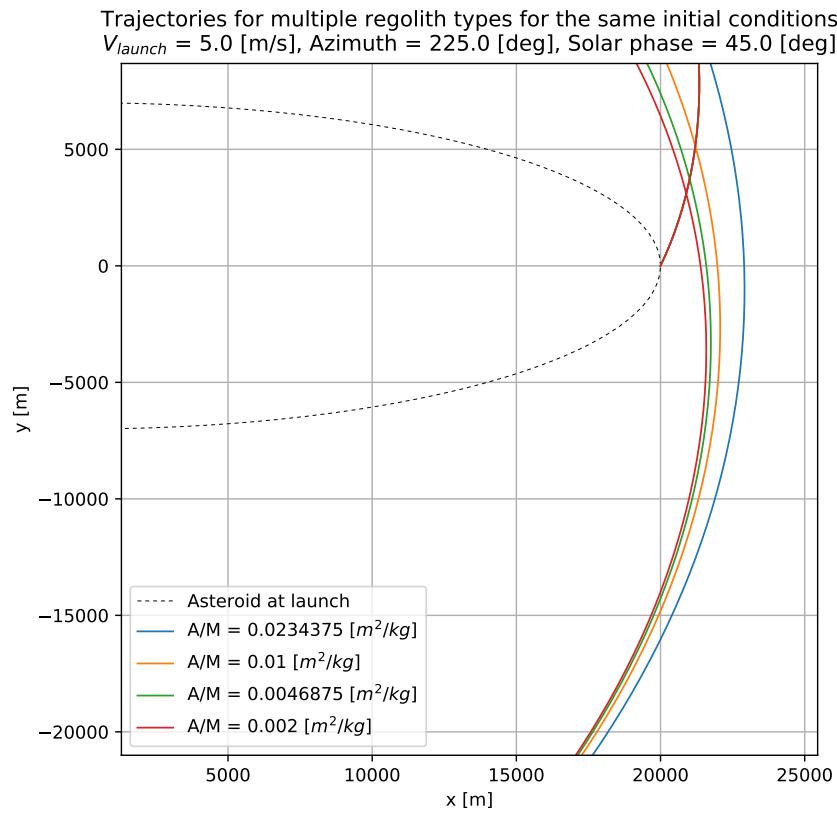


Figure C.13: Zoomed in version of the escape trajectory shown in Figure 7.39a. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

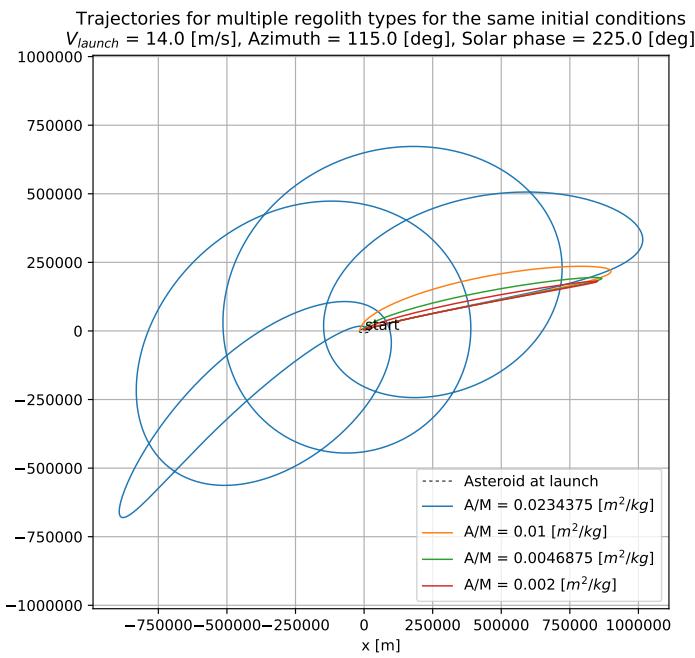


Figure C.14: Re-impact trajectories for all regolith types, launched with the same initial conditions. Particle trajectories are visibly separated from each other because of the Solar perturbations. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

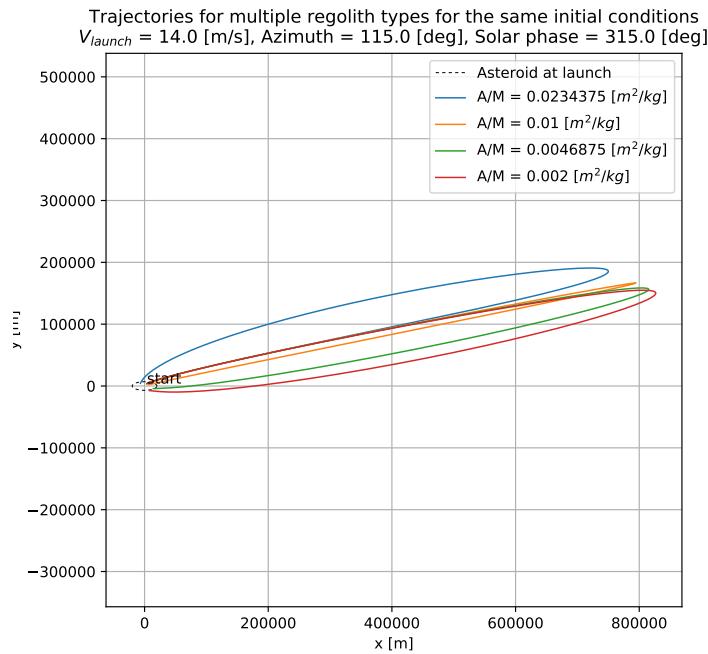


Figure C.15: Re-impact trajectories for all regolith types, launched with the same initial conditions. Particle trajectories are visibly separated from each other because of the Solar perturbations. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

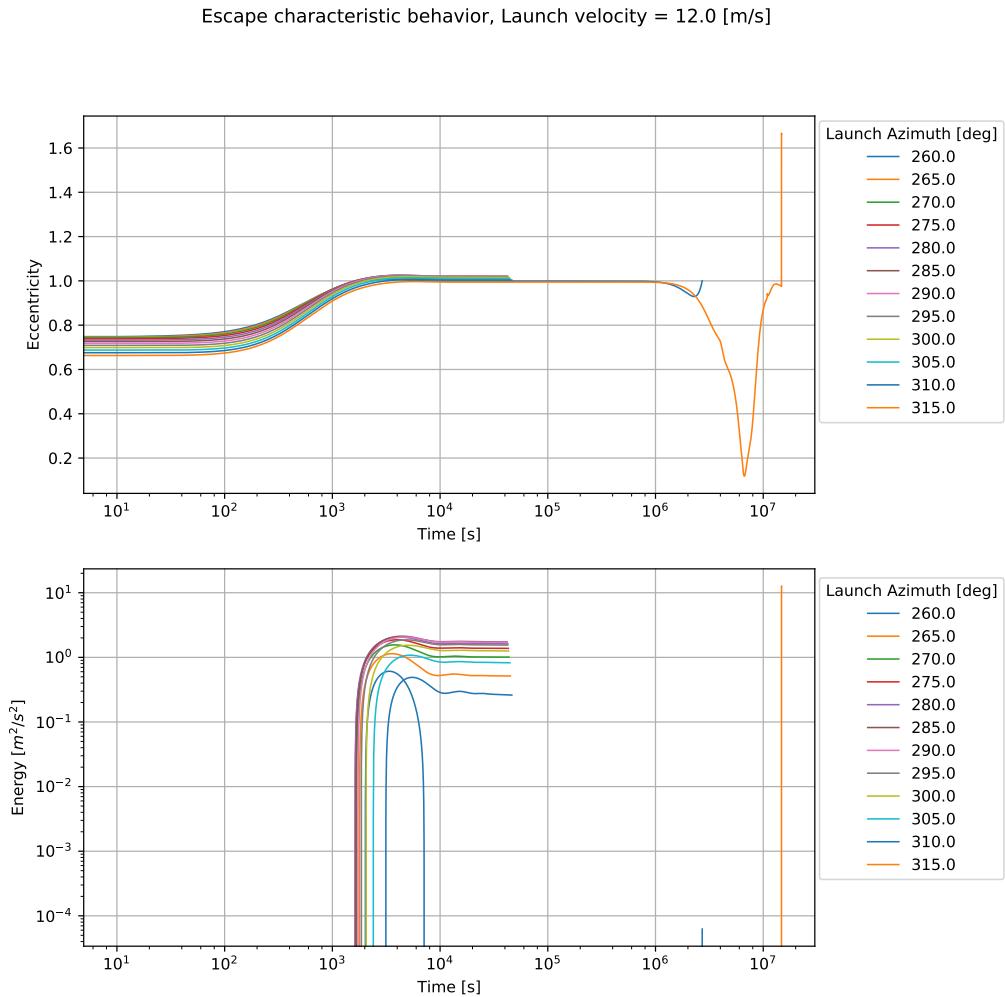


Figure C.16: Plot depicting the variation in energy and eccentricity with time for all escape cases for particle LoGSP-1 launched with a velocity of 12 m/s from Figure 7.50.

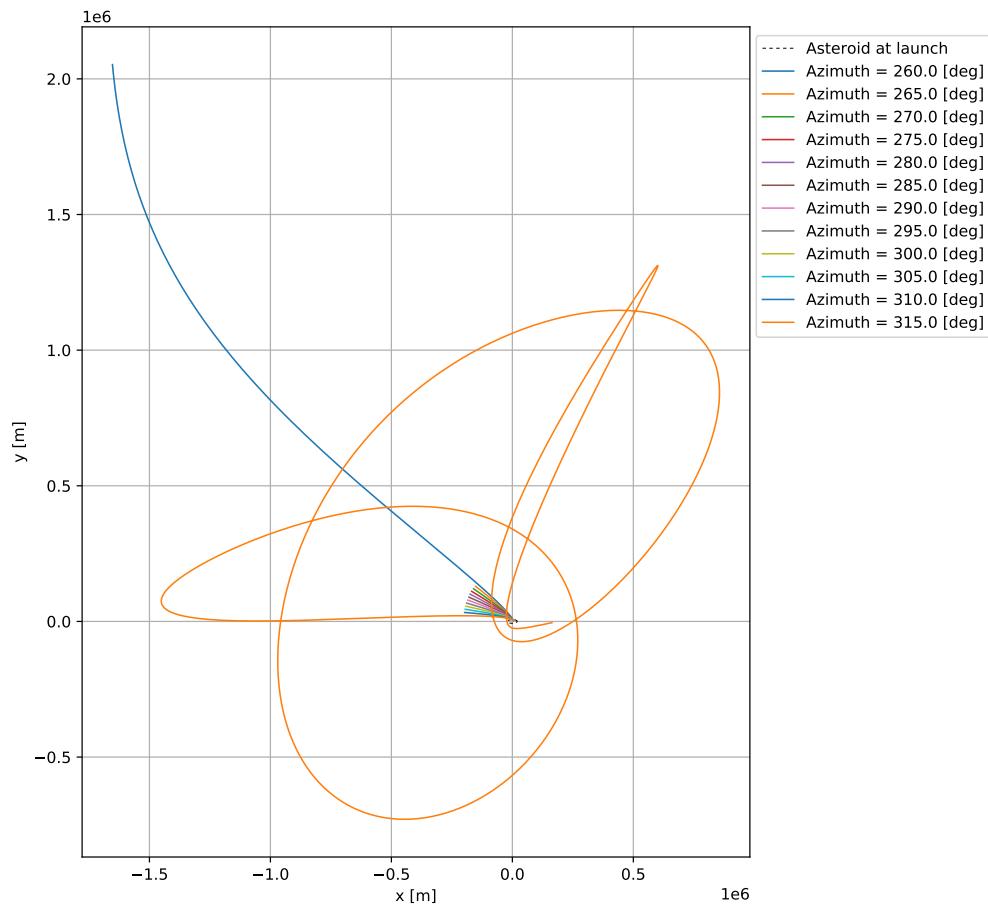


Figure C.17: Plot depicting the trajectory evolution for all escape cases for particle LoGSP-1 launched with a velocity of 12 m/s from Figure 7.50.

Escape characteristic behavior, Launch velocity = 14.0 [m/s]

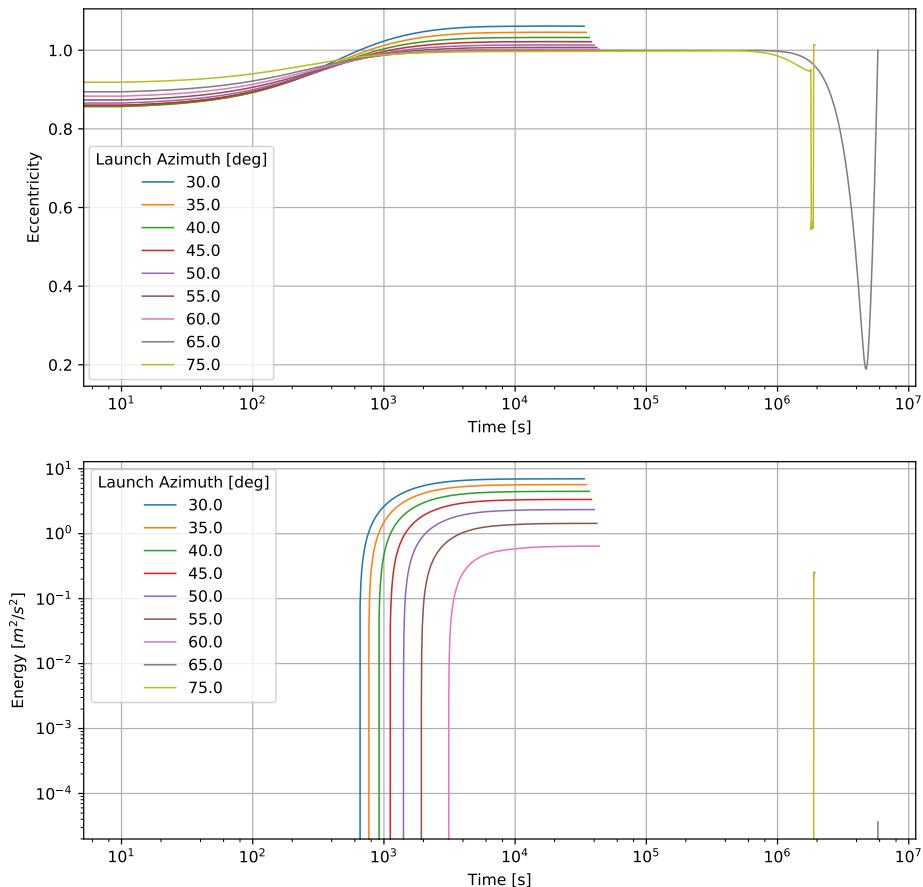


Figure C.18: Plot depicting the variation in energy and eccentricity with time for escape cases, for particle LoGSP-1 launched with a velocity of 14 m/s and azimuth 30° to 100°, from Figure 7.50.

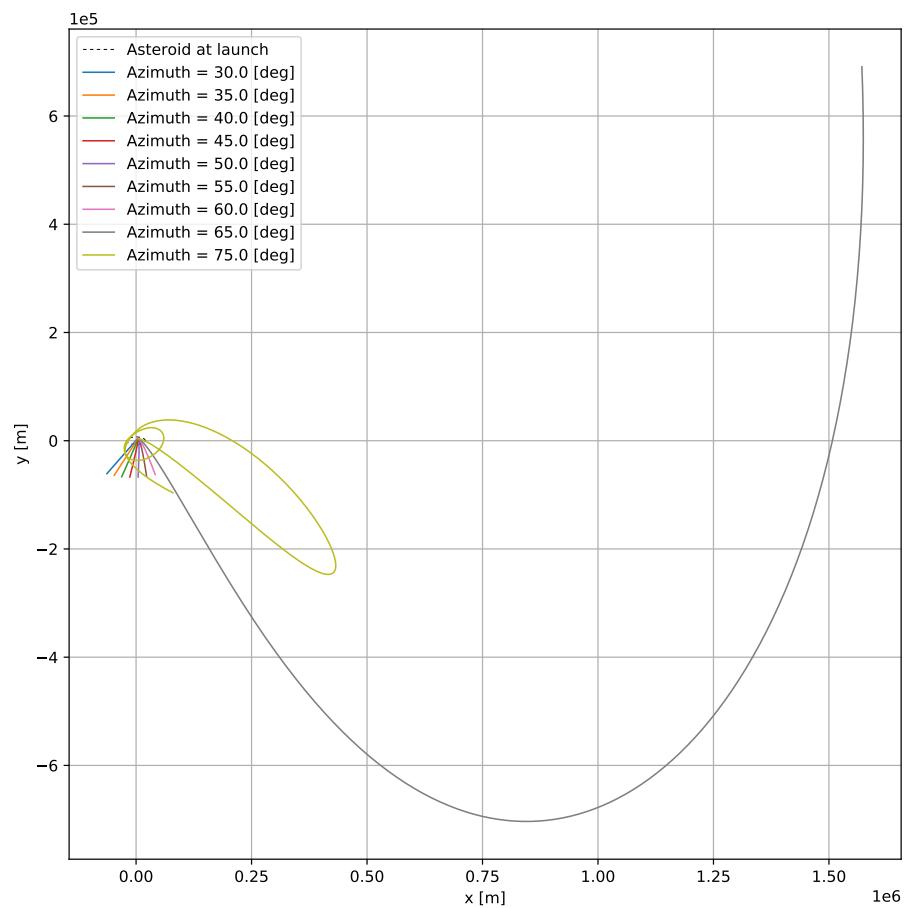


Figure C.19: Plot depicting the trajectory evolution for escape cases, for particle LoGSP-1 launched with a velocity of 14 m/s and azimuth 30° to 100°, from Figure 7.50.

Escape characteristic behavior, Launch velocity = 14.0 [m/s]

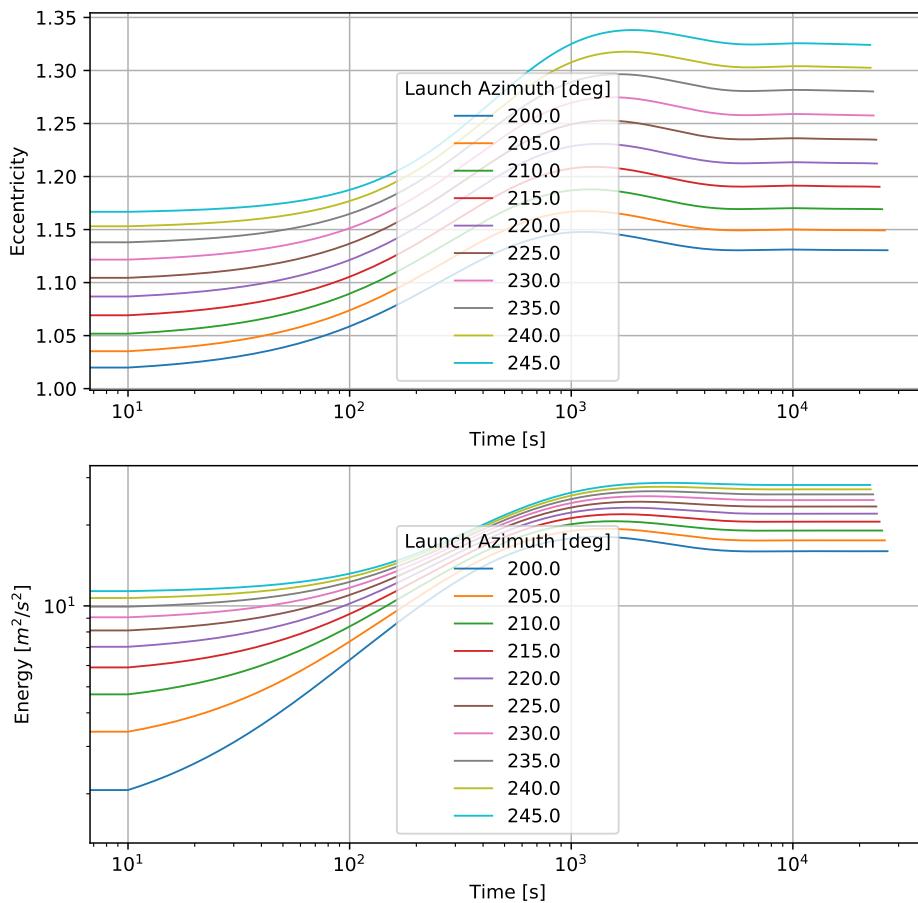


Figure C.20: Plot depicting the variation in energy and eccentricity with time for escape cases, for particle LoGSP-1 launched with a velocity of 14 m/s and azimuth 200° to 250°, from Figure 7.50.

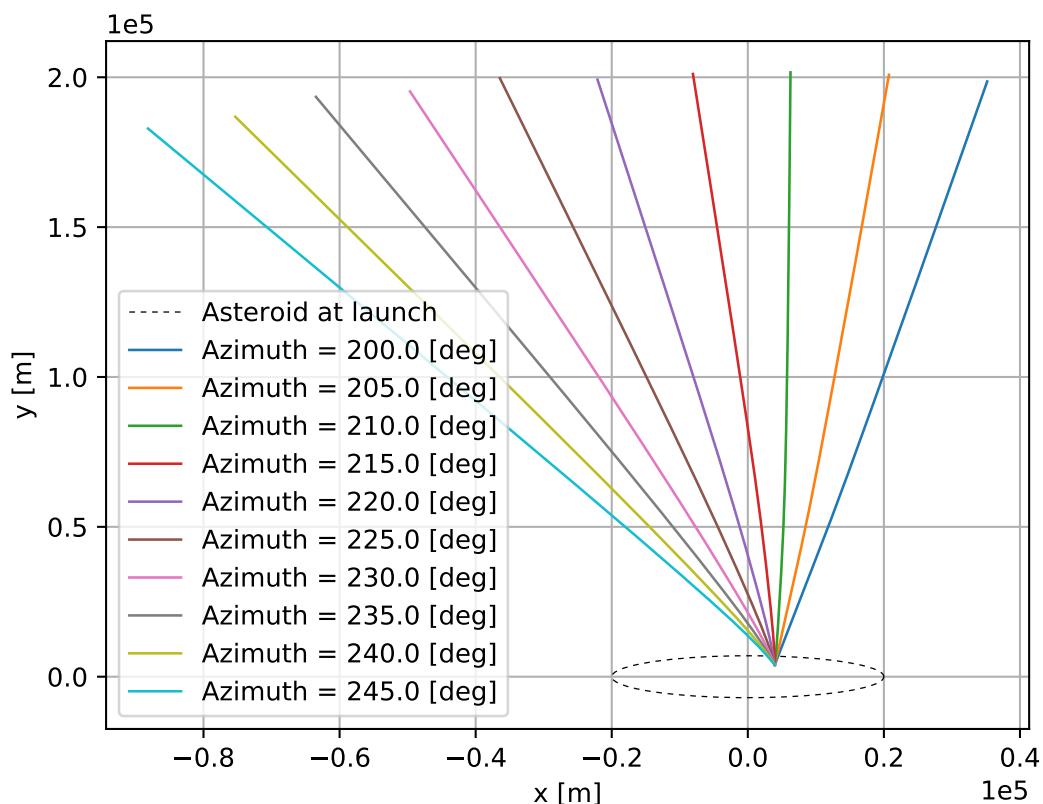


Figure C.21: Plot depicting the trajectory evolution for escape cases, for particle LoGSP-1 launched with a velocity of 14 m/s and azimuth 200° to 250°, from Figure 7.50.

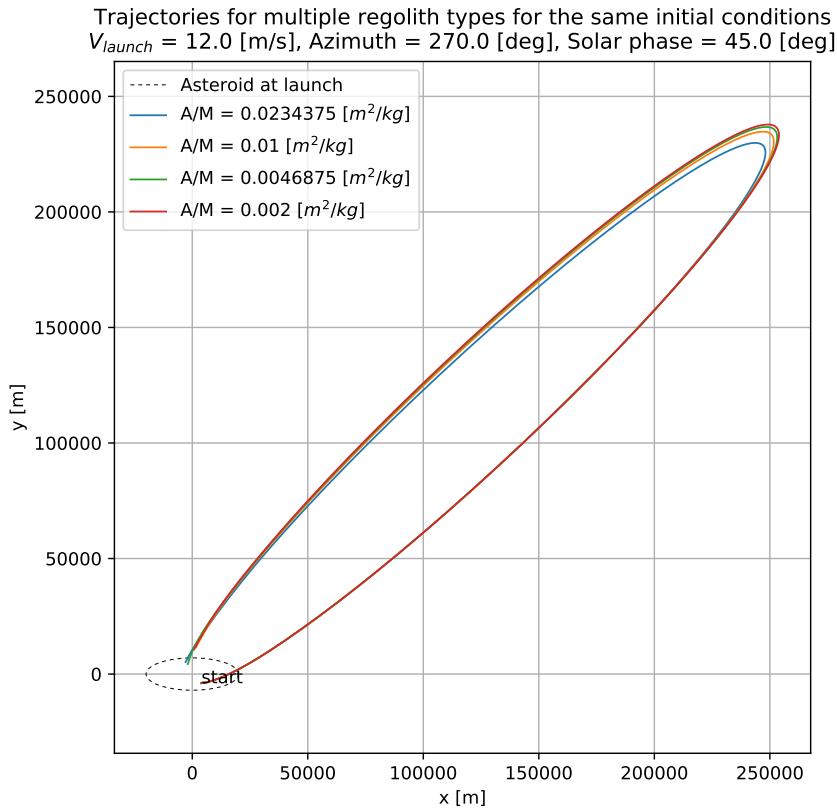


Figure C.22: Re-impact trajectories shown for all regolith types, for launch velocity 12 m/s, azimuth 270° and initial Solar phase angle of 45°. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

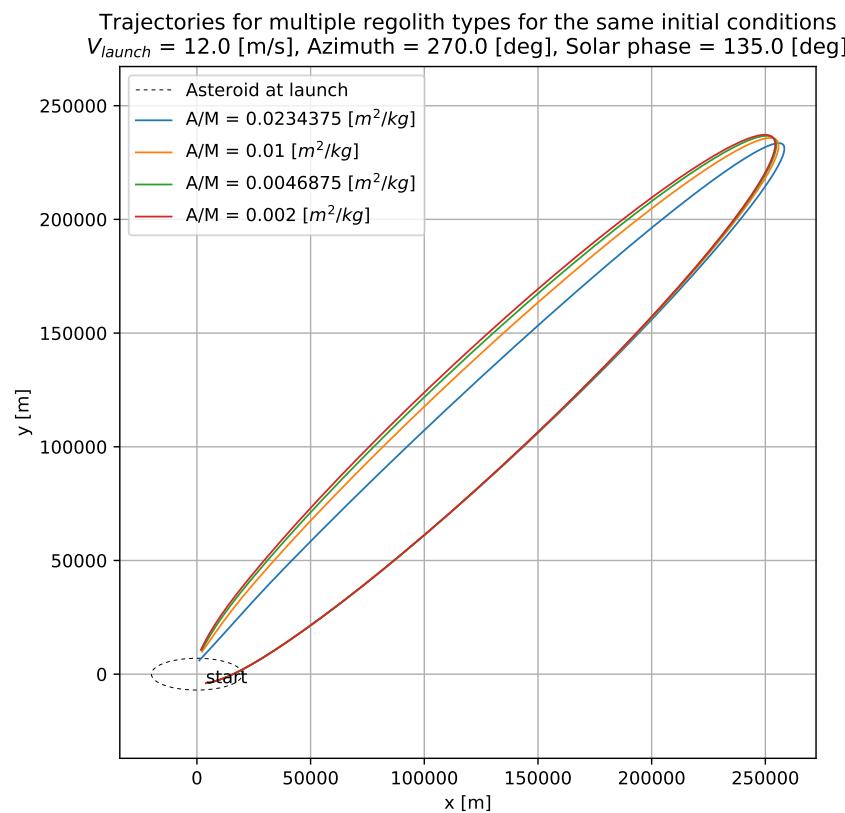


Figure C.23: Re-impact trajectories shown for all regolith types, for launch velocity 12 m/s, azimuth 270° and initial Solar phase angle of 135°. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

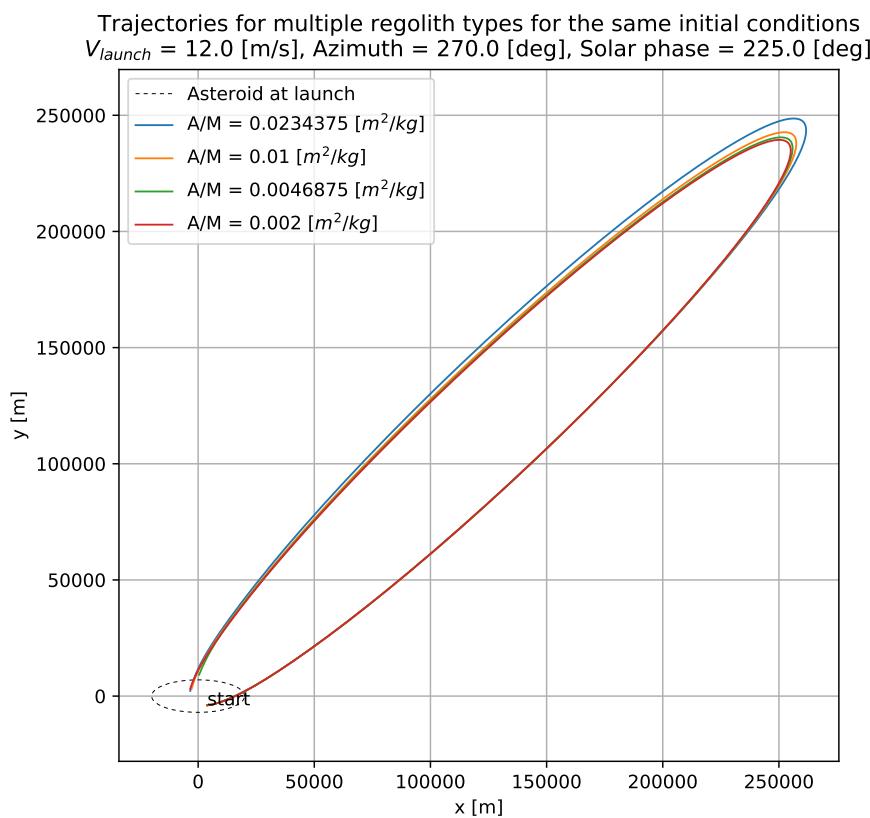


Figure C.24: Re-impact trajectories shown for all regolith types, for launch velocity 12 m/s, azimuth 270° and initial Solar phase angle of 225°. The densities and the particle size for the corresponding area-to-mass ratios can be found in Table 7.1.

Escape characteristic behavior, Launch velocity = 9.0 [m/s]

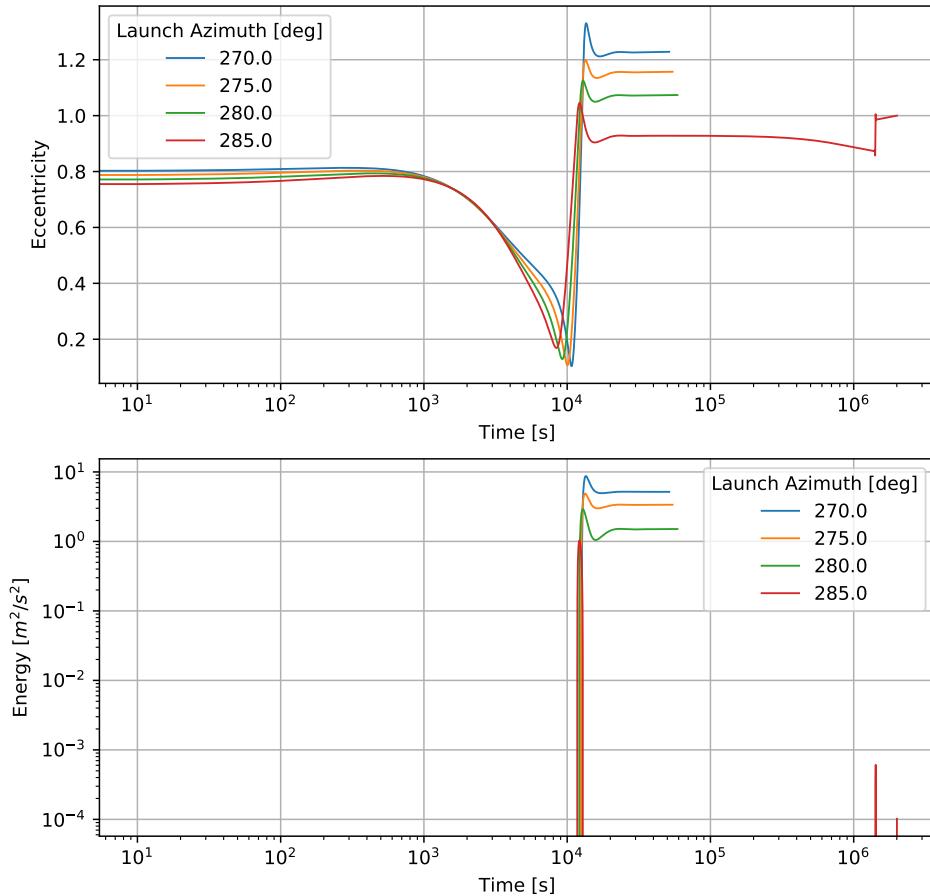


Figure C.25: Plot depicting the variation in energy and eccentricity with time for escape cases, for particle LoGSP-1 launched with a velocity of 9 m/s, from Figure 7.61.

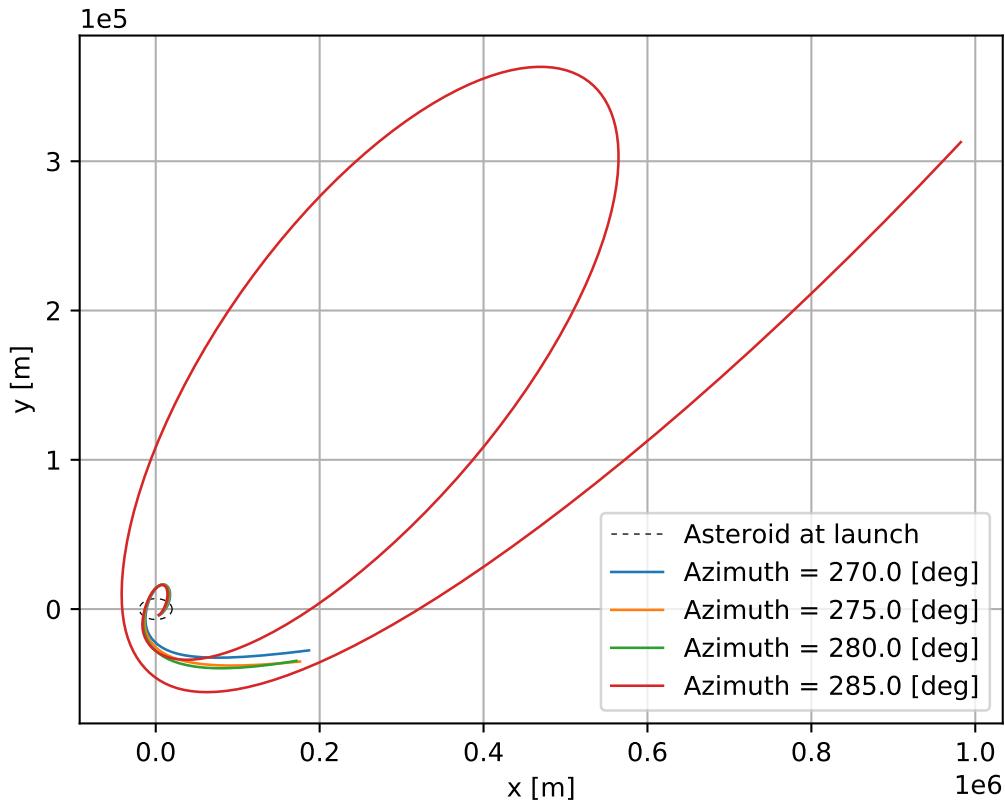


Figure C.26: Plot depicting the trajectory evolution for escape cases, for particle LoGSP-1 launched with a velocity of 9 m/s, from Figure 7.61.

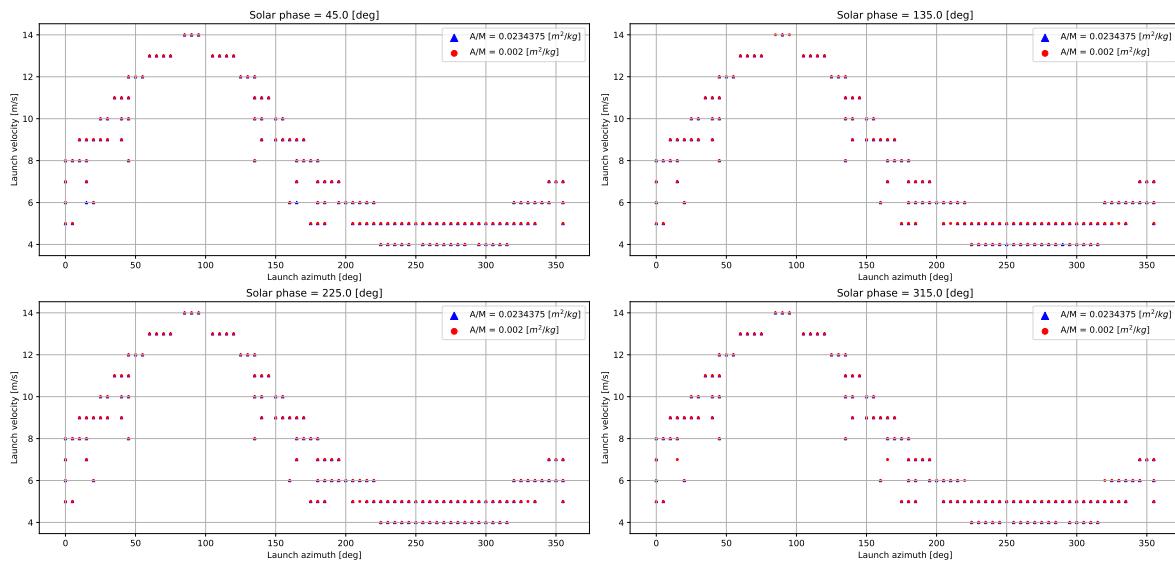


Figure C.27: Plot depicting the initial conditions for particles LoGSP-1 and LoGSP-4 wherein the orbits last longer than one Earth day. The regoliths were launched from the longest edge.

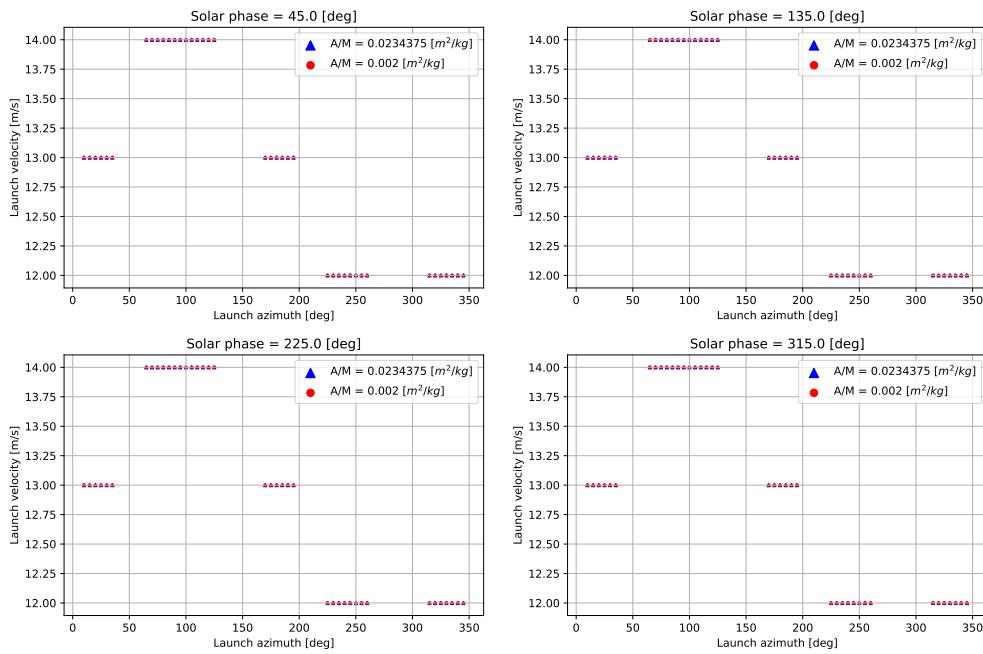


Figure C.28: Plot depicting the initial conditions for particles LoGSP-1 and LoGSP-4 wherein the orbits last longer than one Earth day. The regoliths were launched from the leading edge.

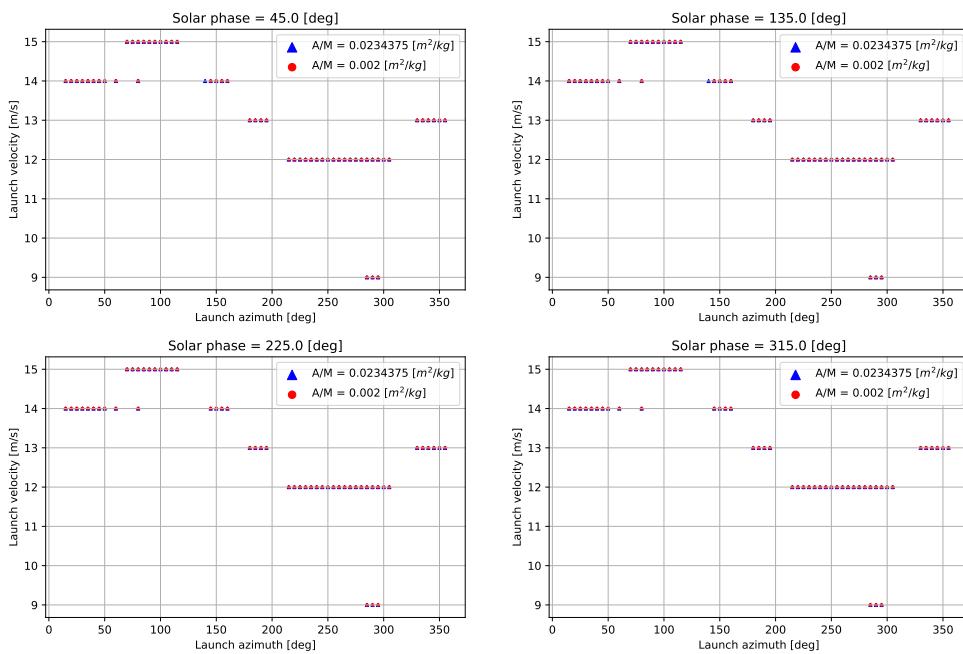


Figure C.29: Plot depicting the initial conditions for particles LoGSP-1 and LoGSP-4 wherein the orbits last longer than one Earth day. The regoliths were launched from the trailing edge.

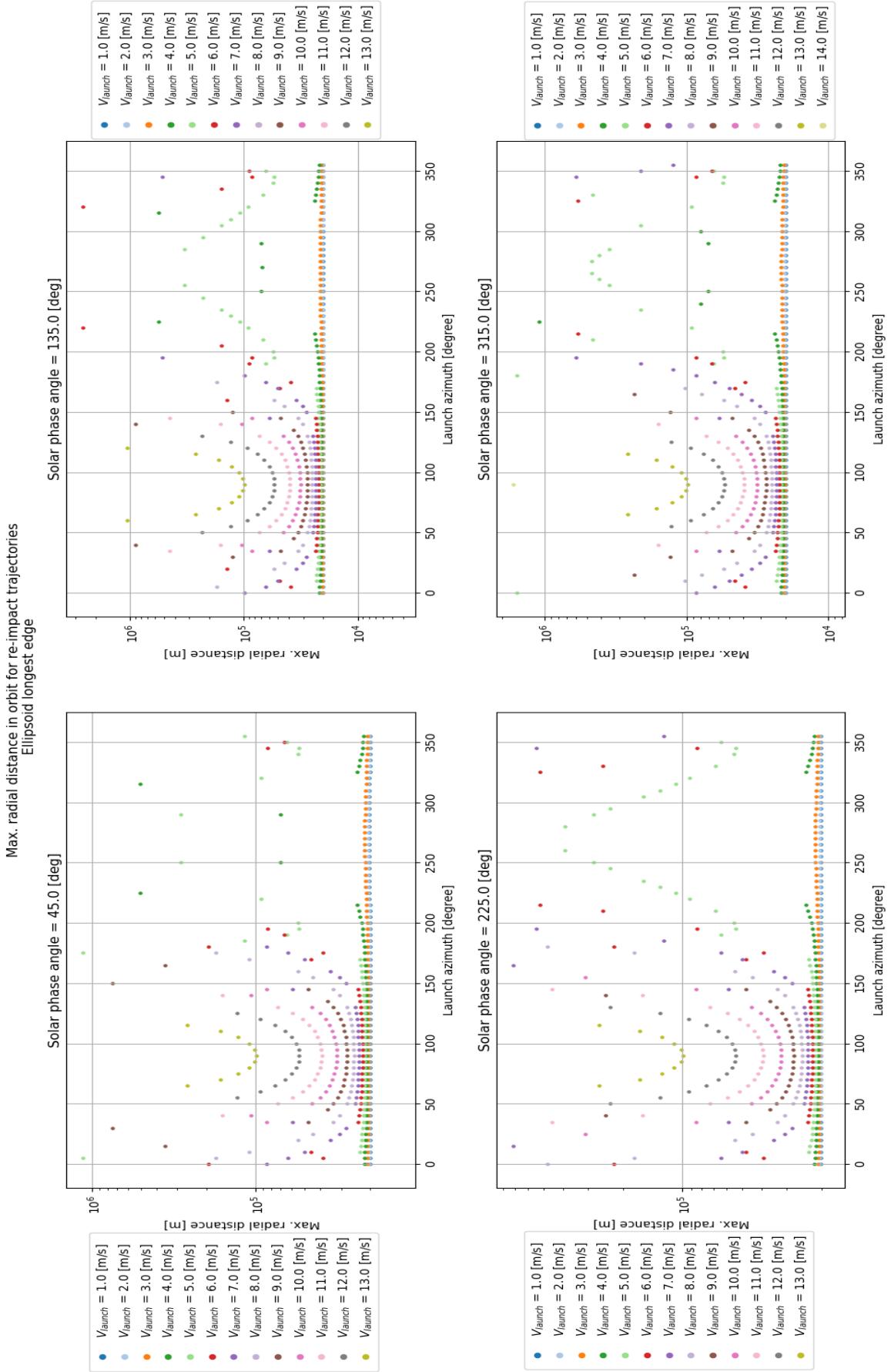
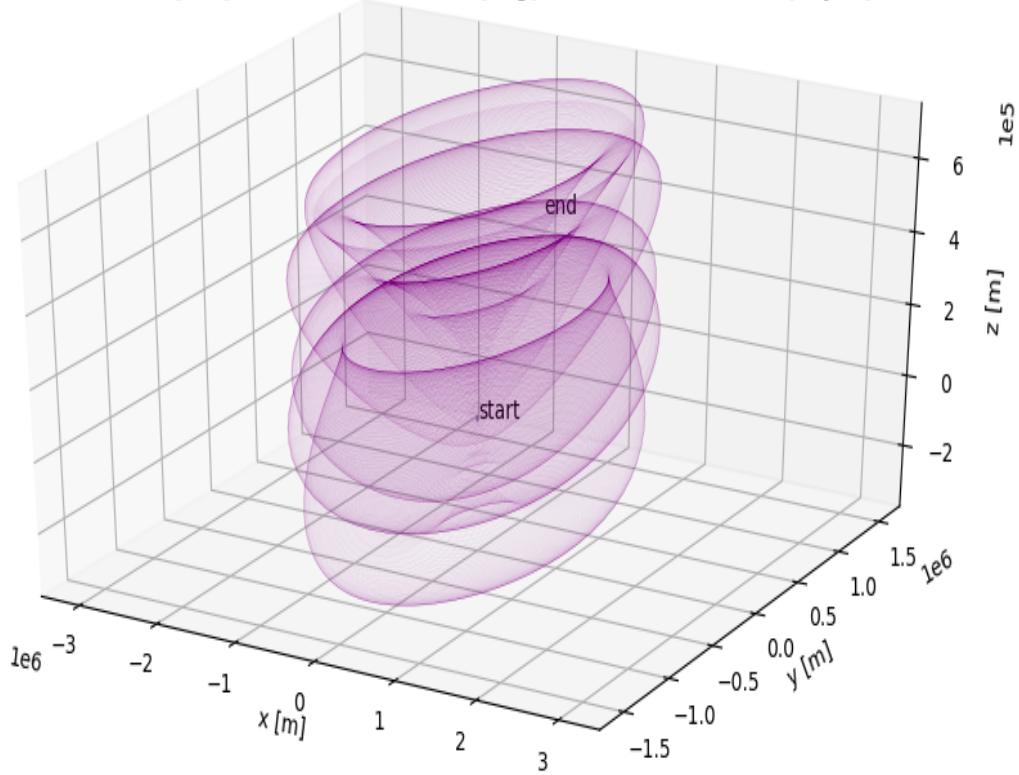


Figure C.30: Maximum radial distance (from the centre of the asteroid) attained by the regolith in orbit for different launch velocities and launch azimuths. The particles were launched from the longest edge of the ellipsoid (asteroid).

Plots are for particle code LoGSP-1 and only for the re-impact scenario.

Particle trajectory around asteroid Eros (Body frame)
 $V_{initial}=10.0[\text{m/s}]$, Launch azimuth=45.0[deg], time=269.996608796[day(s)]



Particle trajectory around asteroid Eros (Inertial frame)
 $V_{initial}=10.0[\text{m/s}]$, Launch azimuth=45.0[deg], time=269.996608796[day(s)]

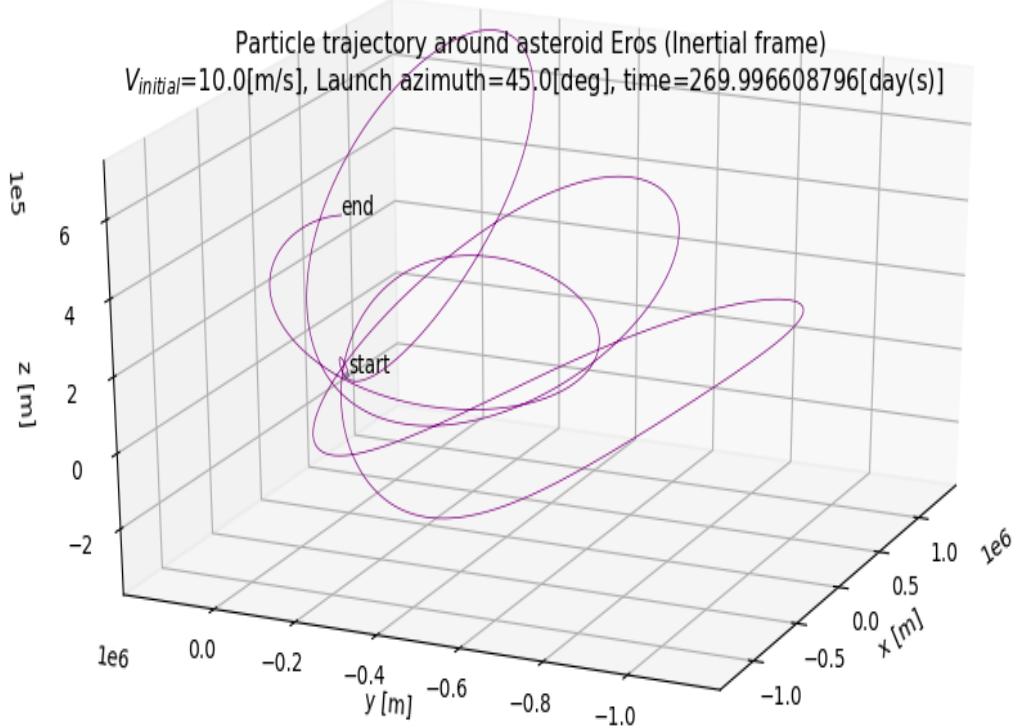


Figure C.31: 3D trajectory of capture regolith for case number 5 in Table 7.2. Particle code LoGSP-1.

Particle trajectory projection around asteroid Eros (Body fixed frame)
 $V_{initial}=10.0[\text{m/s}]$, Launch azimuth=45.0[deg], time=6479.91861111[hrs]

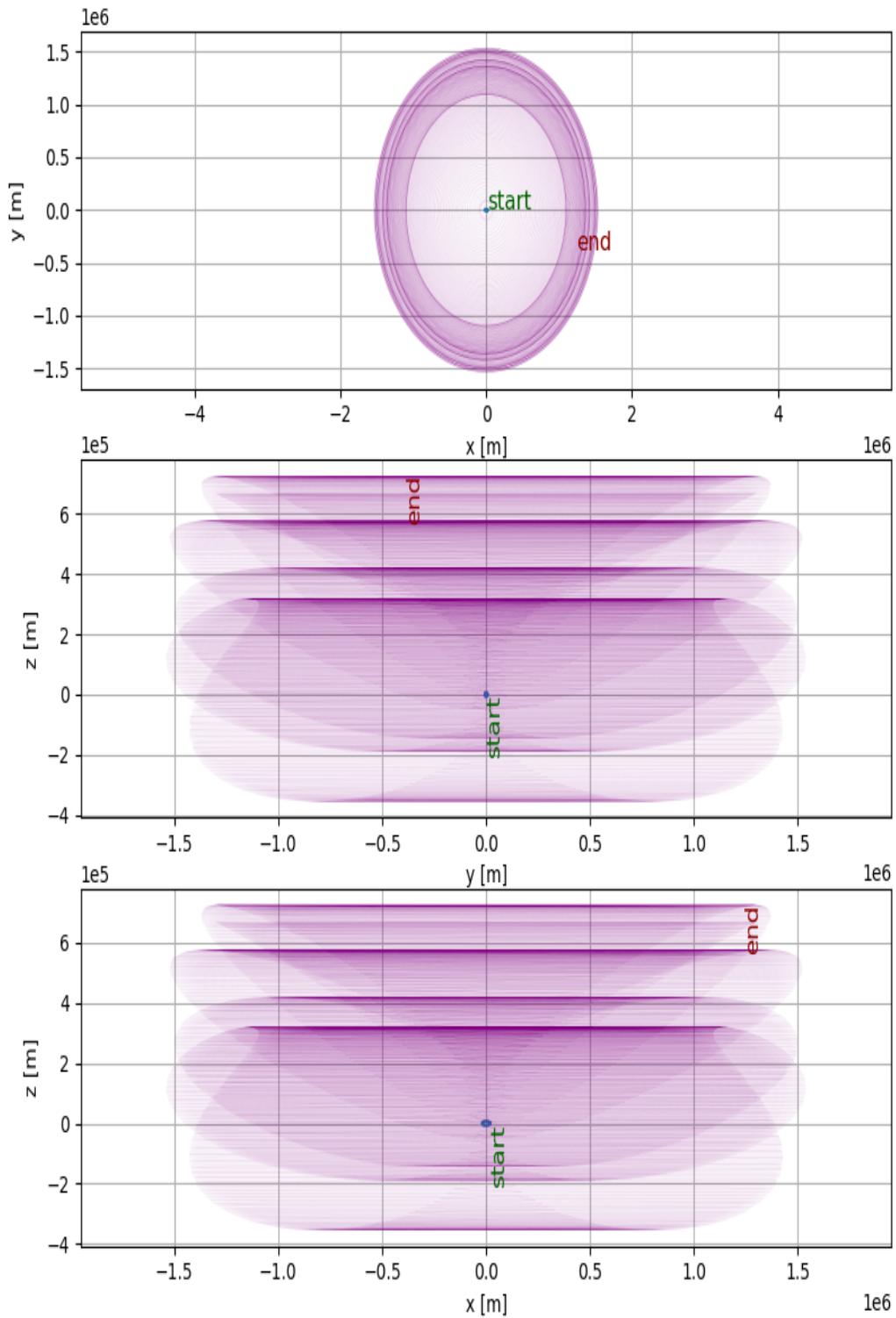


Figure C.32: 2D rotating frame trajectory of capture regolith for case number 5 in Table 7.2. Particle code LoGSP-1.

Particle trajectory around asteroid Eros (Body frame)
 $V_{initial}=8.0[\text{m/s}]$, Launch azimuth=165.0[deg], time=269.998056713[day(s)]

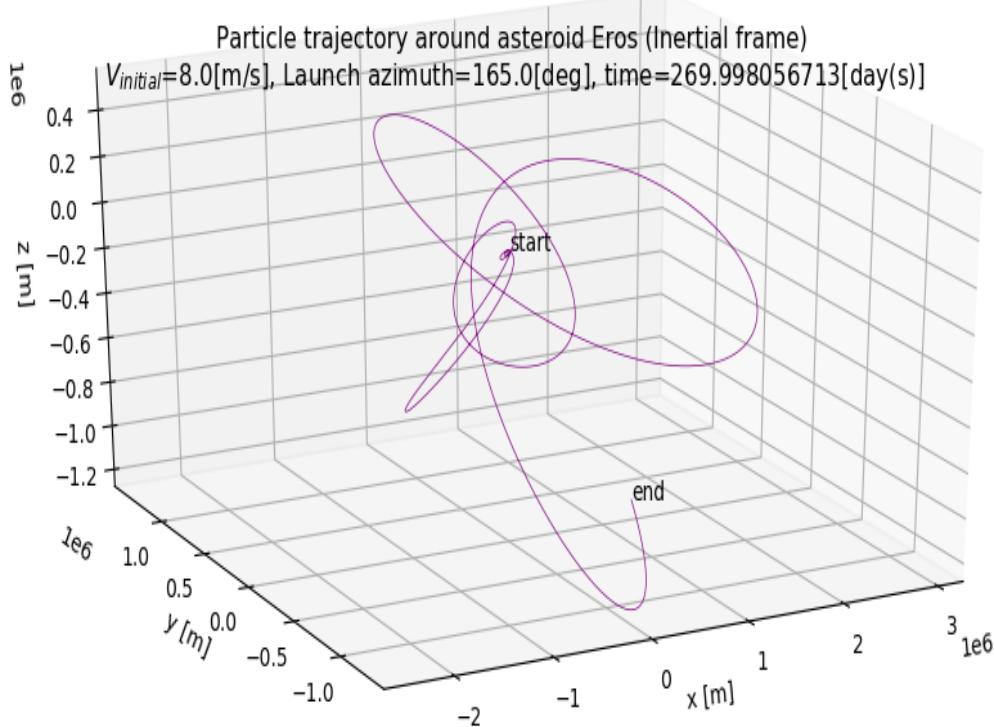
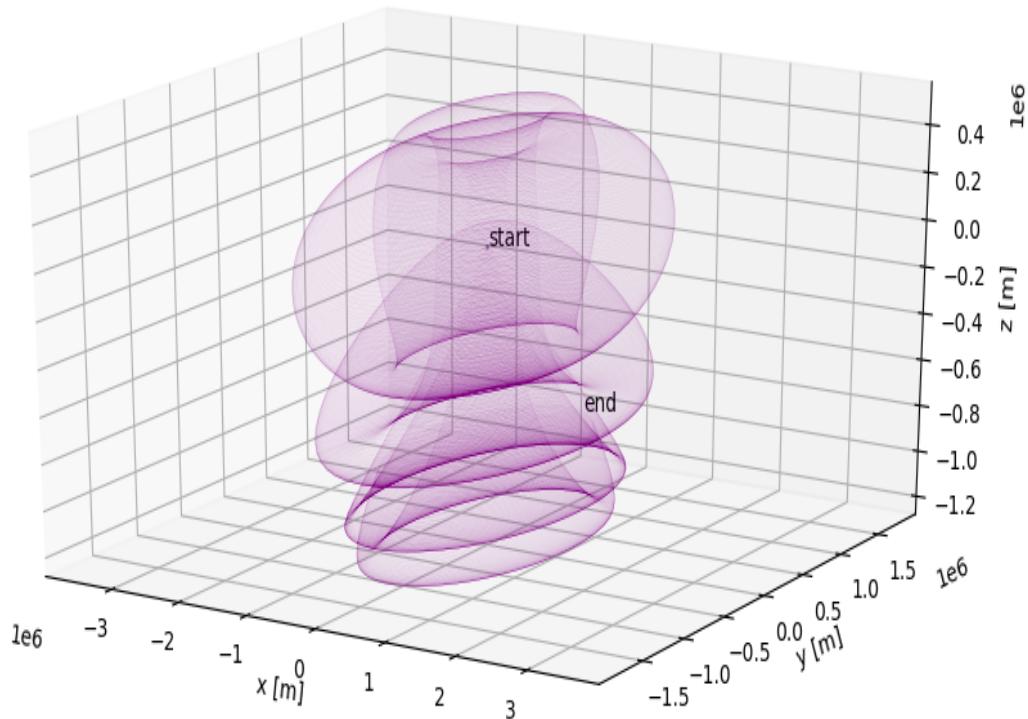


Figure C.33: 3D trajectory of capture regolith for case number 5 in Table 7.2. Particle code LoGSP-1.

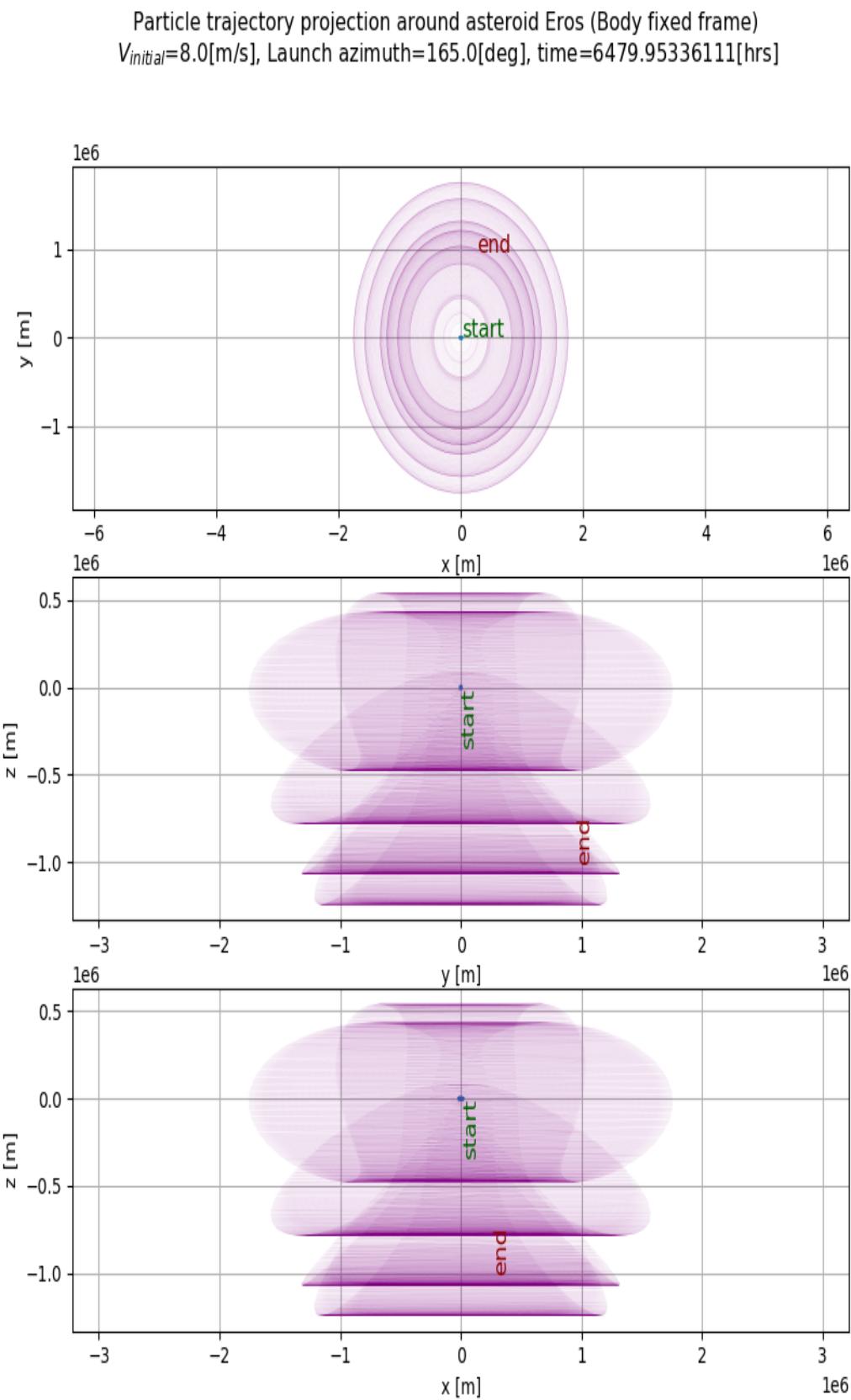


Figure C.34: 2D rotating frame trajectory of capture regolith for case number 8 in Table 7.2. Particle code LoGSP-1.