

Preliminary Design

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1 Introduction

This document outlines and documents the conceptual and preliminary design considerations for pigeon. The conceptual design is the early high-level process to establish over all needs and design goals. The preliminary design is used to determine features of the basic components/subsystems.

The conceptual design is the first and most important phase of the system design and development process [1]. Typically, the appropriate starting point is to identify the problem and the associated definitions of need at a conceptual level [1].

2 Conceptual Design

This section is divided into a few subsections to cover each conceptual design group independent of one another. An early attempt of establishing a foundation describing the primary aircraft components, configuration, and classification is made. By the end of this section the reader should have an understanding of what the objectives are for the aircraft, its classification, and conceptual design features. It is also a primary objective to list other "standard" aircraft of similar classification and configuration to assist in defining realistic baseline requirements.

2.1 Design Goals

This section outlines the overall baseline requirements that the aircraft shall be constrained to. Research has been done to get a general understanding of what the hobbyist's average system consists of. The chosen aircraft are of similar configuration that the pigeon will have. This research is summarized in Table 1.

Table 1: Short list of remote-controlled aircraft of similar configurations and some of their components.

Plane	Wing Span [mm]	Length [mm]	Flying Weight [g]
H-King (PNF) P-51D Moonbeam McSwine	750	648	425
Duraflly Tundra V2	1300	920	1150
H-King (ARF) Savage Bobber	1000	700	650
Avios (PNF) Grand Tundra	1700	1260	—
Duraflly Ugly Stick V2	1100	950	1050
H-King Crusader	1200	1000	990

The goals that the pigeon aim to achieve **shall** be to give a platform for basic avionics research to be conducted on. Starting goals consist of having a basic remote controlled aircraft that has stable flight. Once this has been achieved, the desired goal is to implement autonomous components, research, and further engineering on the aircraft. However, before this can be achieved baseline requirements must be met.

The listed remote-control aircraft can be considered "introductory" as they are budget, lightweight, and do not have a high flight weight (the maximum amount of weight the aircraft can sustain and still become airborne). An average weight of a rc-plane can be range from 1134-1588 grams (2.5-3.5 pounds). From Table 1, a lot of these aircraft come close to carrying double their own weight. Using these aircraft's materials and designs as inspiration should, theoretically at least, allow similar numbers to be met. Therefore, the pigeon **shall** have a weight of no more than 11588 grams with a flying weight of 853 grams. The flight weight is the average of those in Table 1. The maximum weight of any component will most likely be the power supply. Preliminary research of the largest battery used by the aircraft listed above is a 4500 mAh LiPo battery. These batteries have a weight of about 662 grams. That leaves about 191 grams for the other components such as the on board computer, actuators, microcontrollers, cameras, and other sensors.

Preliminary research shows that an average speed of these entry level model planes range from 64 to 97 kph (40 to 60 mph). Considering that this is a research focused aircraft, speed is not a pressing matter; however, it does correlate to the maximum lift force which may need to be in the upper end of the distribution to be able to accommodate increasing demands for weight as new components/sensors are added. As a starting point, the desired baseline speed **shall** to be above 64 kmh.

Flight time for an average rc-plane is 15-20 minutes. Considering more power will be utilized by the added electronics for the sensors, on board computer, and microcontrollers, flight time may be impacted by this. Therefore, the flight time of the pigeon **shall** be no less than 10 minutes.

2.2 Classification

The pigeon is to be a hybrid radio-controlled/autonomous aircraft system. It has no intent of modeling after any other real life aircraft, otherwise known as being a flying model. The radio-controlled system, as any other typical remote controlled aircraft, has a transmitter with the pilot operating it from the ground. The radio-controlled portion will be designed first along with the aircraft to ensure that the system is capable of stable flight.

Once this has been achieved, the autonomous dual to the system will begin to be introduced in phases. These phases, subject to change, include take off, landing, collision avoidance, way point following, etc. These phases, however, do not account for internal engineering efforts such as software design, control design, filtering, etc. These concepts will be flushed out in more detail at a later date.

2.3 Primary Aircraft Components and Configuration

The basic components of the aircraft are: wing, fuselage, horizontal tail, vertical tail, engine, and landing gear. These components are essential in describing the plane. A high level configuration for each is to be described. The terminology convention follows [1]. The wing configuration **shall** be a mono-plane, high wing, rectangular, and fixed wing. The tail configuration **shall** be an aft tail, conventional, and fixed. The propulsion system **shall** be a single-engine placed in front of nose, tractor, and propeller driven. The landing gear configuration **shall** be a fixed tail gear. The fuselage configuration **shall** be an unpressurized cabin where the electronic components will seat. The aircraft **shall** be built of Expanded polypropylene (EPP) is an extremely resilient variety of foam, often used in basic trainers, which take considerable abuse from beginners, PLA plastic, and carbon fiber rods. The subsystem configuration **shall** have a conventional primary control surfaces, fly-by-wire power transmission, battery will be inside the fuselage, and the aircraft **shall** store cameras, sensors, and other electronics on board.

3 Preliminary Design

3.1 Weight Buildup

The total weight is composed of different components of the aircraft.

$$W_{TO} = W_{PL} + W_A + W_B + W_E \quad (1)$$

Which are the total weight, payload weight, battery weight, and the empty weight. This can be rearranged into

$$W_{TO} = \frac{W_{PL} + W_A}{1 - \frac{W_B}{W_{TO}} - \frac{W_E}{W_{TO}}} \quad (2)$$

3.1.1 Payload weight

Payload weight of the UAV is the net carrying capacity of the aircraft. This includes cameras sensors, camera, radar, etc. These values are determined by the specification sheet as well as a safety factor $Sf = 1.5$.

```
Sf      = 1.5;                # Safety factor
w_cam   = 50;                 # Camera weight
w_lidar = 50;                 # Lidar weight

W_pl    = Sf*(w_cam + w_lidar); # Payload weight

disp(W_pl);                    # Display the payload weight

octave> octave> octave> octave> octave> 150
```

3.1.2 Autopilot weight

This is the weight that contributes to the flight operations of the aircraft. In the instance of the pigeon, it consists of both radio systems and autopilot. The mass of the inertial measurement unit (IMU) devices are very small relative to the other autopilot pieces. These include sensors such as altimeter, GPS, gyroscope/accelerometer.

```
w_pc = 46;           # Onboard PC weight
w_mc = 25;           # Microcontroller weight
w_rad = 27.22;       # Radio receiver weight
w_imu = 50;          # IMU weights

W_a = Sf*(w_pc + w_mc + w_rad + w_imu);  # Autopilot weight

disp(W_a);           # Display autopilot weight

octave> octave> octave> octave> octave> octave> 222.33
```

3.1.3 Battery Weight

The pigeon is a battery-driven aircraft. The battery weight is a major contributor to the weight buildup of the aircraft. [1] shows that for a propeller-driven aircraft, its battery weight ratio can be written as shown in Equation 3. g is the gravity constant, η_P is the propeller efficiency, E_D energy density of the battery, R is the desired range, C_L is the cruise lift coefficient, and C_D is the drag due to the aircraft configuration.

$$W_B = 1.05 \left(\frac{g}{\eta_P E_D} \frac{R}{C_L / C_D} \right) \quad (3)$$

The propeller efficiency has typical values ranging from 0.6 through 0.8. For the sake of adding safety factors in the design, the low end of the efficiency shall be used. The range can be determined from Equation 4. V_C denotes the cruising velocity. Average starter RC aircraft velocities range from 64 to 97 kph. Solving for R is simple, and we have defined the flight time to be about 10 minutes. The output of ?? calculates the estimated range.

$$t_C = \frac{R}{V_C} \quad (4)$$

```
R = (10) * (64 * (1000/60));           # min * m/min

disp(R);

octave> 1.0667e+04
```

References

- [1] Mohammad H. Sadraey. *Aircraft Design*. Wiley, September 2012.