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Linear Quadratic Integral Differential Game applied to the Real-time Control of a Quadrotor Experimental setup

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Abstract

This research paper presents a novel approach to quadrotor attitude control that draws on differential game theory. The approach uses a linear quadratic Gaussian (LQG) controller with integral actions. Accurate attitude control is of utmost importance for safe and effective quadrotor flight, particularly in the presence of disturbances. To develop a dependable and effective control system, the motion equations with nonlinearity for the quadrotor's experimental setup are transformed into a continuous-time state-space model through linearization. The model parameters are identified through experimental data, and a two-player approach is employed to determine the attitude control commands. By mini-maximizing a quadratic set of criteria, which is the total of the outputs and disturbances weighted by the amount of control effort, one player minimizes the command while the other generates disturbances. The performance of the proposed approach is evaluated by comparing it to a linear quadratic regulator controller in level flight. The results demonstrate that the proposed approach effectively dissipates disturbances and outperforms linear quadratic regulator controllers, thereby contributing to the development of robust and effective attitude control systems for quadrotors.

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Keywords:

Linear quadratic Gaussian controller, Differential game theory, Quadrotor, Continuous state-space model, 3-degree-of-freedom experimental platform, Attitude Control Optimization, Robust disturbance rejection.

1. Introduction

The investigation, strategic operations, optical sensing, entertainment, and farming are all used by quadrotors in today's society [1]. Various subsystems of the quadrotor control system are responsible for the quadrotor's perfor-

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mance, including attitude, altitude, and position. Maintaining the desired attitude outputs is essential for quadrotor attitude control, particularly in the presence of sudden disturbances. This can be achieved by controlling the rotor rotational speeds[2]. Consequently, there is a growing body of research focusing on the development of automatic control strategies for quadrotors to effectively manage disturbances and maintain desired attitudes. The quadrotor attitude is controlled by a Proportional Integral Derivative (PID) controller in [3, 4]. During disturbances, however, this controller has not effectively achieved the control objectives. Model-based approaches to controller design are utilized to resolve this issue [5, 6]. Quadrotor attitude models and disturbances determine the direction of the best control commands using these controllers.

The literature has proposed numerous models-based controllers to provide a faster control algorithm when dealing with modeling errors and reducing disturbances. An intelligent controller, a robust controller, a nonlinear controller, and an optimal controller are some of these types of controllers.

Various control approaches based on intelligent logic, such as machine learning [7], evolutionary computation [8], iterative learning [9], reinforcement learning [10], and fuzzy logic [11], have been extensively employed for quadrotor attitude regulation. Numerous nonlinear control techniques have been proposed to regulate orientation angles of quadrotors, including Synergetic Control [12], Sliding Mode Control (SMC) [13], and Feedback Linearization (FBL) [14]. Robust control methods such as H_∞ [15, 16] and μ -synthesis [17] have also been utilized to stabilize the attitudes of quadrotors under conditions of extreme uncertainty and worst-case scenarios. Quadrotors have also been controlled using optimal controllers, such as the Linear Quadratic Gaussian (LQG) [18] and the Linear Quadratic Regulator (LQR) [19]. These controllers minimize a quadratic criterion by considering both regulation performance and control effort to provide optimal feedback gains. The Linear Quadratic Regulator Differential Game (LQR-DG) is a robust and optimal control strategy that has been extensively utilized in controlling various nonlinear and complex systems, including quadrotors. This approach employs a linear system model to control its outputs while minimizing a cost function through mini-maximization. The LQR-DG approach has been demonstrated to offer excellent regulation performance and control effort while being robust to external disturbances. For instance, it has been successfully applied to control a ship controller, showcasing its effectiveness and versatility in handling complex control problems [20, 21]. By means of an analytical pursuit-evasion process, the LQR-DG control approach generates precise and optimized control commands. This unique feature sets the LQR-DG controller apart from other conventional optimal control techniques and enables one player to track the most appropriate control command while the other creates disturbances.

This study proposes a new controller for quadrotors based on the LQG technique with integral action, using differential game theory to optimize control performance in the presence of disturbance. Attitude control in the presence of disturbance is essential for ensuring safe and effective quadrotor flight. In order to devise this control approach, a continuous state-space model for a quadrotor's experimental setup is formulated by linearizing the system's equations of motion, which are inherently nonlinear, and identifying its parameters based on experimental data. This approach involves transforming the nonlinear dynamics into a linear system that can be analyzed more readily. The proposed Linear Quadratic Regulator Differential Game (LQR-DG) controller offers an optimal control strategy for generating efficient control commands for quadrotors. The approach employs a two-player system, wherein one player optimizes the command while the other creates disturbances through mini-maximization of a quadratic criterion. This technique not only ensures efficient control but also offers a robust and versatile approach to handle complex control problems. The proposed approach is evaluated against a linear quadratic regulator controller in level flight. The results show that the LQIG-DG controller effectively dissipates disturbances and outperforms the linear quadratic regulator controller. This study presents an important contribution to the development of robust and effective attitude control systems for quadrotors.

The following sections of this research paper provide a comprehensive analysis of the proposed control approach. Section 2 defines the problem, while sections 3 and ?? offer detailed derivations of the dynamics model and the estimation problem for the quadrotor's experimental platform, respectively. Section 4 explicates the proposed LQIG-DG controller architecture. The effectiveness of the controller is evaluated in section 5, which is followed by the conclusion in section 6.

2. Problem Statement

A complex nonlinear system describes the experimental platform of the quadrotor depicted in Figure 1. During its rotation, the quadrotor can adjust its pitch, roll, and yaw angles. An Inertial Measurement Unit (IMU) is used to measure the acceleration and angular velocities along the three orthogonal axes, which are affected by noise. To estimate the states of the quadrotor, including its angular velocities and Euler angles, a nonlinear filter is utilized. A graphical representation of the LQIG-DG controller structure is depicted in Figure 2. The estimated states allow for the stabilization of the quadrotor setup.



Figure 1: 3DoF setup of the quadrotor.

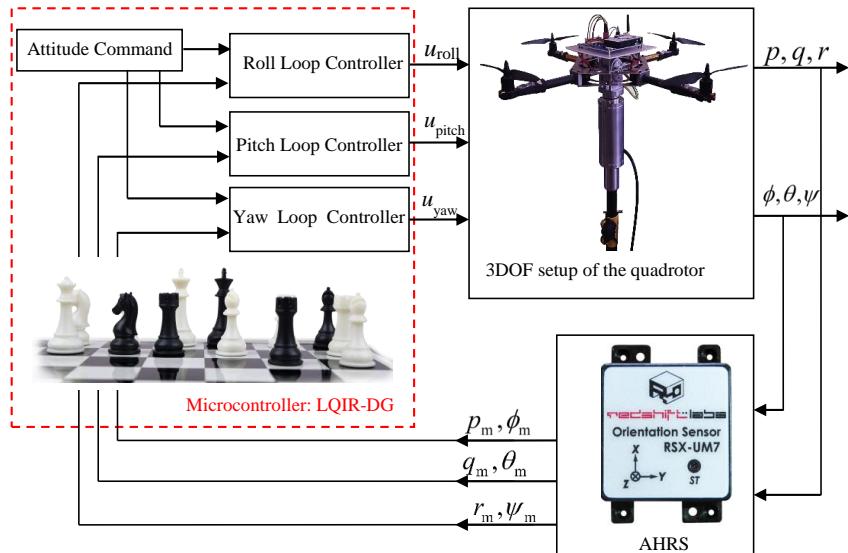


Figure 2: Structure of the LQIG-DG Controller Illustrated in a Block Diagram.

3. Nonlinear Dynamic Model for the Quadrotor Platform

This research paper presents an analysis of the three degrees of freedom for the quadrotor. Furthermore, a state-space model is developed to represent the nonlinear attitude dynamics. A linearization technique is subsequently applied to the model, enabling it to be used as a LQIR-DG control model.

3.1. Quadrotor Configuration and Attitude Dynamics Model

Figure 3 illustrates the quadrotor schematic, depicting four rotors with angular velocity Ω_r rotating around the z_B axis in the body coordinate system. The rotational direction of Rotor 1 and Rotor 3 in the counterclockwise direction counteracts the yawing moment, whereas the clockwise rotation of Rotors 2 and 4 counteracts the yawing moment.

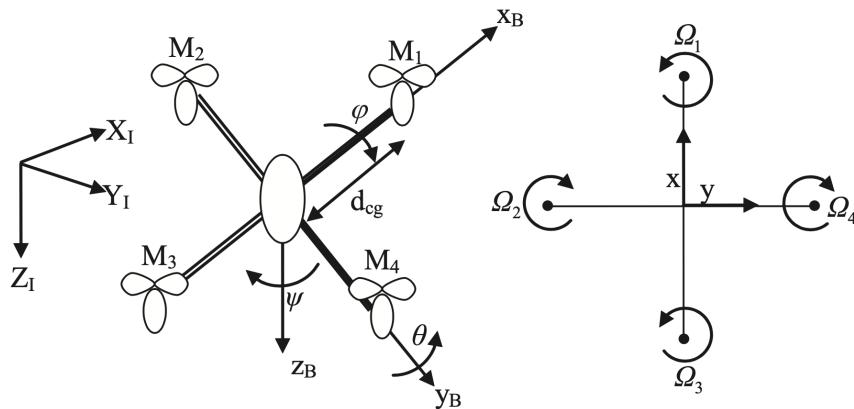


Figure 3: Configuration of the quadrotor.

3.2. Dynamic Modeling of the Quadrotor Platorm

The dynamic model of the Quadrotor Platorm is an essential component for control design. In this section, the quadrotor dynamic model is derived using the Newton-Euler method [22, 23].

$$\dot{p} = \frac{I_{yy} - I_{zz}}{I_{xx}} qr + q \frac{I_{rotor}}{I_{xx}} \Omega_r + \frac{u_{roll}}{I_{xx}} + \frac{d_{roll}}{I_{xx}} \quad (1)$$

$$\dot{q} = \frac{I_{zz} - I_{xx}}{I_{yy}} rp + p \frac{I_{rotor}}{I_{xx}} \Omega_r + \frac{u_{pitch}}{I_{yy}} + \frac{d_{pitch}}{I_{yy}} \quad (2)$$

$$\dot{r} = \frac{I_{xx} - I_{yy}}{I_{zz}} pq + \frac{u_{yaw}}{I_{zz}} + \frac{d_{yaw}}{I_{zz}} \quad (3)$$

The variables (p, q, r) represent the rotational velocities, while the variables d_{roll} , d_{pitch} , and d_{yaw} denote the disturbances produced in x_B , y_B , and z_B , respectively. Additionally, I_{xx} , I_{yy} , and I_{zz} are the principal moments of inertia, and I_{rotor} is the rotor inertia about its axis. Euler angle rates are also determined from angular body rates:

$$\begin{bmatrix} \dot{\phi} \\ \dot{\theta} \\ \dot{\psi} \end{bmatrix} = \begin{bmatrix} 1 & \sin(\phi) \tan(\theta) & \cos(\phi) \tan(\theta) \\ 0 & \cos(\phi) & -\sin(\phi) \\ 0 & \sin(\phi)/\cos(\theta) & \cos(\phi)/\cos(\theta) \end{bmatrix} \begin{bmatrix} p \\ q \\ r \end{bmatrix} \quad (4)$$

The quadrotor system has three degrees of freedom: roll, pitch, and yaw. The rotational motion is described by the angles (ϕ, θ, ψ) , which represent roll, pitch, and yaw, respectively. The residual rotor velocity, denoted by ω_r , is calculated as follows:

$$\Omega_r = -\Omega_1 + \Omega_2 - \Omega_3 + \Omega_4 \quad (5)$$

3.3. Innovative Control Strategies for Quadrotor

The control inputs u_{roll} , u_{pitch} , and u_{yaw} correspond to the moments generated by the quadrotor's rotors along the roll, pitch, and yaw axes, respectively, and are defined as follows:

$$u_{\text{roll}} = b d_{\text{cg}} (\Omega_2^2 - \Omega_4^2) \quad (6)$$

$$u_{\text{pitch}} = b d_{\text{cg}} (\Omega_1^2 - \Omega_3^2) \quad (7)$$

$$u_{\text{yaw}} = d (\Omega_1^2 - \Omega_2^2 + \Omega_3^2 - \Omega_4^2) \quad (8)$$

The quadrotor's drag and thrust coefficients are denoted by d and b , respectively. The distance between the rotors and the gravity center is represented by d_{cg} . The angular velocity commands can be computed as follows:

$$\Omega_{c,1}^2 = \Omega_{\text{mean}}^2 + \frac{1}{2b d_{\text{cg}}} u_{\text{pitch}} + \frac{1}{4d} u_{\text{yaw}} \quad (9)$$

$$\Omega_{c,2}^2 = \Omega_{\text{mean}}^2 + \frac{1}{2b d_{\text{cg}}} u_{\text{roll}} - \frac{1}{4d} u_{\text{yaw}} \quad (10)$$

$$\Omega_{c,3}^2 = \Omega_{\text{mean}}^2 - \frac{1}{2b d_{\text{cg}}} u_{\text{pitch}} + \frac{1}{4d} u_{\text{yaw}} \quad (11)$$

$$\Omega_{c,4}^2 = \Omega_{\text{mean}}^2 - \frac{1}{2b d_{\text{cg}}} u_{\text{roll}} - \frac{1}{4d} u_{\text{yaw}} \quad (12)$$

The nominal angular velocities of the rotors are denoted by Ω_{mean} .

3.4. State-Space Representation of Quadrotor Dynamics

The formulation of a state-space model is crucial for the development of advanced control strategies. In this context, a state-space representation of the quadrotor system is provided in order to facilitate the design of control algorithms.

In order for the quadrotor to be controlled, it can be modelled in state-space by introducing the following variables: $x_1 = p$, $x_2 = q$, $x_3 = r$, $x_4 = \phi$, $x_5 = \theta$, and $x_6 = \psi$.

$$\dot{x}_1 = \Gamma_1 x_2 x_3 + \Gamma_2 x_2 \Omega_r + \Gamma_3 u_{\text{roll}} + \Gamma_3 d_{\text{roll}} \quad (13)$$

$$\dot{x}_2 = \Gamma_4 x_1 x_3 - \Gamma_5 x_1 \Omega_r + \Gamma_6 u_{\text{pitch}} + \Gamma_6 d_{\text{pitch}} \quad (14)$$

$$\dot{x}_3 = \Gamma_7 x_1 x_2 + \Gamma_8 u_{\text{yaw}} + \Gamma_8 d_{\text{yaw}} \quad (15)$$

$$\dot{x}_4 = x_1 + (x_2 \sin(x_4) + x_3 \cos(x_4)) \tan(x_5) \quad (16)$$

$$\dot{x}_5 = x_2 \cos(x_4) - x_3 \sin(x_4) \quad (17)$$

$$\dot{x}_6 = (x_2 \sin(x_4) + x_3 \cos(x_4)) / \cos(x_5) \quad (18)$$

Furthermore, to facilitate the analysis, $\Gamma_i (i = 1, \dots, 8)$ is introduced, which represents a set of coefficients related to the quadrotor's physical properties and external factors.

$$\begin{aligned} \Gamma_1 &= \frac{I_{yy} - I_{zz}}{I_{xx}}, & \Gamma_2 &= \frac{I_{\text{rotor}}}{I_{xx}}, & \Gamma_3 &= \frac{1}{I_{xx}} \\ \Gamma_4 &= \frac{I_{zz} - I_{xx}}{I_{yy}}, & \Gamma_5 &= \frac{I_{\text{rotor}}}{I_{xx}}, & \Gamma_6 &= \frac{1}{I_{yy}} \\ \Gamma_7 &= \frac{I_{xx} - I_{yy}}{I_{zz}}, & \Gamma_8 &= \frac{1}{I_{zz}} \end{aligned} \quad (19)$$

The attitude states and gyro measurements are represented as vectors, denoted by \mathbf{z} and \mathbf{m} , respectively. The attitude state vector is defined as follows:

$$\mathbf{z} = [p_m \ q_m \ r_m \ \phi_m \ \theta_m \ \psi_m]^T \quad (20)$$

Note that the vector is transposed using the superscript T to indicate that it is a row vector.

3.5. Linearization of the Nonlinear Quadrotor Model

The application of a continuous-time linear model facilitates the efficient control of the quadrotor. The linear state-space model can be represented as follows:

$$\dot{\mathbf{x}}(t) = \mathbf{A} \mathbf{x}(t) + \mathbf{B} \mathbf{u}(t) + \mathbf{B}_d \mathbf{d}(t) \quad (21)$$

The system's input and disturbance matrices, denoted as \mathbf{B} and \mathbf{B}_d respectively, with \mathbf{d} representing the disturbance, are part of the system's dynamics, represented by the matrix \mathbf{A} . The measurement equation is given by

$$\mathbf{z}(t) = \mathbf{x}(t) \quad (22)$$

The differential equations presented in Equations (13)-(18) allow for the development of a corresponding linear dynamic model for the quadrotor platform at its equilibrium points ($\mathbf{x}_e \neq 0$ and $\mathbf{u}_e \neq 0$). This model can be expressed as follows:

$$\begin{aligned} \dot{\mathbf{x}} = & \begin{bmatrix} \dot{\mathbf{x}}_{\text{roll}} \\ \dot{\mathbf{x}}_{\text{pitch}} \\ \dot{\mathbf{x}}_{\text{yaw}} \end{bmatrix} = \begin{bmatrix} \mathbf{A}_{\text{roll}} & \mathbf{0} & \mathbf{0} \\ \mathbf{0} & \mathbf{A}_{\text{pitch}} & \mathbf{0} \\ \mathbf{0} & \mathbf{0} & \mathbf{A}_{\text{yaw}} \end{bmatrix} \begin{bmatrix} \mathbf{x}_{\text{roll}} \\ \mathbf{x}_{\text{pitch}} \\ \mathbf{x}_{\text{yaw}} \end{bmatrix} \\ & + \begin{bmatrix} \mathbf{B}_{\text{roll}} & \mathbf{0} & \mathbf{0} \\ \mathbf{0} & \mathbf{B}_{\text{pitch}} & \mathbf{0} \\ \mathbf{0} & \mathbf{0} & \mathbf{B}_{\text{yaw}} \end{bmatrix} \begin{bmatrix} \mathbf{u}_{\text{roll}} \\ \mathbf{u}_{\text{pitch}} \\ \mathbf{u}_{\text{yaw}} \end{bmatrix} \\ & + \begin{bmatrix} \mathbf{B}_{\text{roll}} & \mathbf{0} & \mathbf{0} \\ \mathbf{0} & \mathbf{B}_{\text{pitch}} & \mathbf{0} \\ \mathbf{0} & \mathbf{0} & \mathbf{B}_{\text{yaw}} \end{bmatrix} \begin{bmatrix} \mathbf{d}_{\text{roll}} \\ \mathbf{d}_{\text{pitch}} \\ \mathbf{d}_{\text{yaw}} \end{bmatrix} \end{aligned} \quad (23)$$

where $\mathbf{x}_{\text{roll}} = [p \ \phi]^T$, $\mathbf{x}_{\text{pitch}} = [q \ \theta]^T$, and $\mathbf{x}_{\text{yaw}} = [r \ \psi]^T$.

In addition, the input matrices and state are shown as

$$\mathbf{A}_{\text{roll}} = \mathbf{A}_{\text{pitch}} = \mathbf{A}_{\text{yaw}} = \begin{bmatrix} 0 & 0 \\ 1 & 0 \end{bmatrix} \quad (24)$$

$$\mathbf{B}_{\text{roll}} = \begin{bmatrix} 1 \\ I_{xx} \\ 0 \end{bmatrix}; \quad \mathbf{B}_{\text{pitch}} = \begin{bmatrix} 1 \\ I_{yy} \\ 0 \end{bmatrix}; \quad \mathbf{B}_{\text{yaw}} = \begin{bmatrix} 1 \\ I_{zz} \\ 0 \end{bmatrix} \quad (25)$$

3.6. Identification of the Setup Parameters

This section presents an optimization technique for estimating the model parameters (Γ) of the 3DOF experimental setup from experimental data. The technique is based on the Nonlinear Least Squares (NLS) method, which is widely used in parameter estimation problems. The NLS algorithm utilizes the trust-region reflective least squares (TRRLS) method to iteratively find the values of the model parameters. The goal is to minimize a cost function, which is based on the sum of squares between the input/output signals provided by the simulation model and the experimental ones.

The optimization process involves finding a vector \mathbf{F} that minimizes the cost function. This is achieved by iteratively updating the values of the model parameters until convergence is achieved. The NLS method is particularly

useful for problems where the model is nonlinear and the measurement noise is known. The approach is based on a least squares problem, where the objective is to find the values of Γ that minimize the sum of squares function [24]:

$$\min_{\Gamma_i} \left(\| e(\Gamma_i) \|^2 \right) = \min_{\Gamma_i} \left(\sum_{i=1}^n (y - \hat{y})^2 \right) \quad (26)$$

where y and \hat{y} are the experimental and simulated output signals, when the same input signals are applied ones. The structure of the proposed identification approach is illustrated in Figure 4.

In summary, the NLS optimization technique is an effective approach for estimating the model parameters of the 3DOF experimental setup from experimental data. The technique utilizes the TRRLS method to iteratively update the values of the model parameters, with the goal of minimizing the difference between the simulation model and the experimental data.

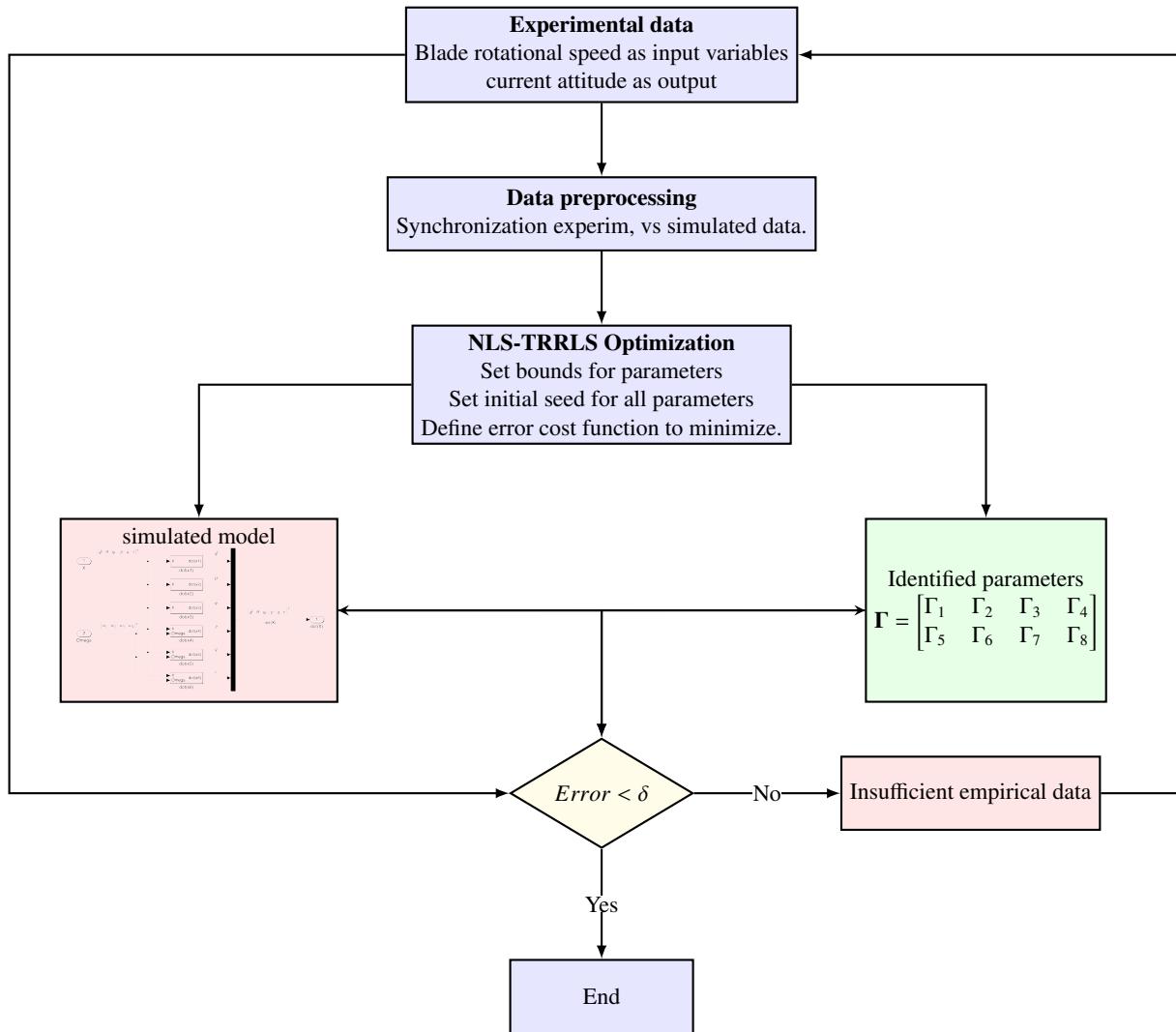


Figure 4: Structure of TRRLS identification approach.

4. Optimal Control Strategy Formulation for a Nonlinear System

The LQR-DG controller is augmented with an integral action to eliminate steady-state errors in desired following. The first stage of the controller design process involves developing the augmented state space representation of the linear quadrotor model to incorporate it into the overall control architecture. Later on, the LQR-DG controller design methodology is introduced to generate optimal control signals for the experimental quadrotor platform.

4.1. Augmented State Space Development

To incorporate the integral action into the controller architecture, the augmented state variables are specified as follows:

$$\mathbf{x}_{\mathbf{a}_i} = \begin{bmatrix} \mathbf{x}_i & \int \mathbf{x}_i \end{bmatrix}^T \quad (27)$$

The dynamics model of the quadrotor, as expressed by Eq. (21), is reformulated into an augmented state-space model that incorporates the roll (ϕ), pitch (θ), and yaw (ψ) angles. The augmented state-space model provides a comprehensive description of the quadrotor's dynamic behavior, enabling the design of effective control strategies. The ensuing state-space model that integrates the augmented state variables is mathematically represented as follows:

$$\dot{\mathbf{x}}_a(t) = \mathbf{A}_a \mathbf{x}_a(t) + \mathbf{B}_a \mathbf{u}(t) + \mathbf{B}_{d_a} \mathbf{d}(t) \quad (28)$$

\mathbf{A}_a and \mathbf{B}_a , which are defined as follows:

$$\mathbf{A}_a = \begin{bmatrix} \mathbf{A} & \mathbf{0} \\ \mathbf{I} & \mathbf{0} \end{bmatrix} \quad (29)$$

$$\mathbf{B}_a = \mathbf{B}_{d_a} = \begin{bmatrix} \mathbf{B} \\ \mathbf{0} \end{bmatrix} \quad (30)$$

In the aforementioned expression, the notation \mathbf{I} denotes the identity matrix.

4.2. LQIR-DG Control Scheme with Integral Action for Quadrotor

In accordance with the principles of differential game theory, the LQIR-DG controller is a robust and optimal methodology. This controller involves two fundamental players, one of which is responsible for determining the optimal control command, while the other strives to generate the worst possible disturbance. The primary objective of the former player is to minimize a cost function, whereas the latter player seeks to maximize it. Thus, the quadratic cost function equation can be expressed using min-max operators, as follows:

$$\min_u \max_d J(\mathbf{x}_{\mathbf{a}_i}, u_i, d_i) = J(\mathbf{x}_{\mathbf{a}_i}, u_i^*, d_i^*) = \min_u \max_d \int_0^{t_f} \left(\mathbf{x}_{\mathbf{a}_i}^T \mathbf{Q}_i \mathbf{x}_{\mathbf{a}_i} + u_i^T R u_i - d_i^T R_d d_i \right) dt \quad (31)$$

In the aforementioned equation, the matrices \mathbf{Q}_i is a symmetric positive definite matrix, while R and R_d are symmetric nonnegative definite matrices. The final time is denoted by t_f . To address this issue, the connection between the general optimal problem and the LQIR problem is established [25]. As a result, the optimal control input for each control loop can be computed using the following formula:

$$u_i(t) = -\mathbf{K}_i(t) \mathbf{x}_{\mathbf{a}_i}(t) \quad (32)$$

$$d_i(t) = \mathbf{K}_{d_i}(t) \mathbf{x}_{\mathbf{a}_i}(t) \quad (33)$$

where \mathbf{K}_{d_i} and \mathbf{K}_i are a time-varying gain, defined as follows:

$$\mathbf{K}_i = R^{-1} \mathbf{B}_{a_i}^T \mathbf{P}_{a_i}(t) \quad (34)$$

$$\mathbf{K}_{d_i} = R_d^{-1} \mathbf{B}_{d_i}^T \mathbf{P}_{d_i}(t) \quad (35)$$

where $\mathbf{P}_{a_i}(t)$ and $\mathbf{P}_{a_{d_i}}(t)$ satisfy

$$\dot{\mathbf{P}}_{a_i}(t) = -\mathbf{A}_a^T \mathbf{P}_{a_i}(t) - \mathbf{P}_{a_i}(t) \mathbf{A}_a - \mathbf{Q}_i + \mathbf{P}_{a_i}(t) \mathbf{S}_{a_i}(t) \mathbf{P}_{a_i}(t) + \mathbf{P}_{a_i}(t) \mathbf{S}_{a_{d_i}}(t) \mathbf{P}_{a_{d_i}}(t) \quad (36)$$

$$\dot{\mathbf{P}}_{a_{d_i}}(t) = -\mathbf{A}_a^T \mathbf{P}_{a_{d_i}}(t) - \mathbf{P}_{a_{d_i}}(t) \mathbf{A}_a - \mathbf{Q}_i + \mathbf{P}_{a_{d_i}}(t) \mathbf{S}_{a_{d_i}}(t) \mathbf{P}_{a_{d_i}}(t) + \mathbf{P}_{a_{d_i}}(t) \mathbf{S}_{a_i}(t) \mathbf{P}_{a_i}(t) \quad (37)$$

In the current investigation, the feedback control law is developed utilizing the steady-state values of the previously mentioned equations as t_f approaches infinity. The matrices \mathbf{S}_{a_i} and $\mathbf{S}_{a_{d_i}}$ are determined by utilizing the matrices \mathbf{B}_{a_i} and $\mathbf{B}_{a_{d_i}}$ along with the symmetric nonnegative definite matrices R and R_d . Specifically, \mathbf{S}_{a_i} is defined as $\mathbf{B}_{a_i} R^{-1} \mathbf{B}_{a_i}^T$, while $\mathbf{S}_{a_{d_i}}$ is defined as $\mathbf{B}_{a_{d_i}} R_d^{-1} \mathbf{B}_{a_{d_i}}^T$.

5. Experimental Findings and Analysis

The current study presents an investigation of the efficacy of the LQIR-DG control strategy applied to the roll, pitch, and yaw control loops of an experimental quadrotor platform. To ensure optimal and robust performance, the controller parameters are tuned using the results of numerical simulations. The performance of the LQIR-DG controller is compared to that of other control strategies, including LQR, PID, and LQIR. The quadrotor's parameters are listed in Table 1.

Furthermore, the weighting parameters of the LQIR-DG controller are provided in Table 2.

Table 1: The Parameter of the Quadrotor

Parameter	Value	Unit
I_{xx}	0.02839	$\text{kg} \cdot \text{m}^2$
I_{yy}	0.03066	$\text{kg} \cdot \text{m}^2$
I_{zz}	0.0439	$\text{kg} \cdot \text{m}^2$
I_{rotor}	4.4398×10^{-5}	$\text{kg} \cdot \text{m}^2$
b	3.13×10^{-5}	$\text{N} \cdot \text{sec}^2 / \text{rad}^2$
d	3.2×10^{-6}	$\text{N} \cdot \text{m} \cdot \text{sec}^2 / \text{rad}^2$
Ω_{mean}	3000	rpm
d_{cg}	0.2	m

Table 2: The Parameters of the LQIR-DG Controller

Control Loop	Weight	Value
Roll	\mathbf{Q}_{roll}	$\text{diag}([0.02, 65.96, 83.04, 0.00])$
Pitch	$\mathbf{Q}_{\text{pitch}}$	$\text{diag}([435.01, 262.60, 262.60, 0.00])$
Yaw	\mathbf{Q}_{yaw}	$\text{diag}([4e-4, 0.00, 0.133, 0])$
-	R	1
-	R_d	1.2764

5.1. Identification of the 3DoF experimental setup model

As described in Section 3.4, the parameters of the quadrotor setup, denoted by $\Gamma_i (i = 1, \dots, 8)$, must be identified using the NRS algorithm. The NLS-TRRLS algorithm is implemented in Matlab R2022b® to increase the accuracy

of parameter identification. Three scenarios, as outlined in Table 3, are considered to ensure comprehensive parameter identification. When the stopping condition of the NLS algorithm is met, the optimal values of the quadrotor parameters are computed and presented in Table 4. The intelligent movement of the parameters during the optimization process is illustrated in Figure 4.

In the first scenario, shown in Figure 5, the quadrotor is rotated about only one axis (roll, pitch, or yaw axes) to identify the parameters Γ_3 , Γ_6 , and Γ_8 . In the second scenario, as illustrated in Figure 6, the parameters Γ_2 and Γ_5 are estimated by allowing the experimental setup to freely rotate around its roll and pitch axes. Finally, in the last scenario, depicted in Figure 7, the parameters Γ_1 , Γ_4 , and Γ_7 of the UAV model are identified by rotating the quadrotor setup around three axes. These results confirm the consistency between the outputs of the simulation results for the quadrotor model and the actual experimental setup.

Table 3: Scenarios for identification of quadrotor model.

Scenario	Description	Initial Conditions		angular velocity Commands			
I	Roll free	38		2000	2000	2000	3400
	Pitch free	-15		3700	2000	2000	2000
	Yaw free	-75		2000	3300	2000	3300
II	Roll and Pitch free		8 -5	1700	3800	2400	1700
III	Roll, Pitch, and Yaw free	8	-3 -146	1700	3800	2400	1700

Table 4: True values of the quadrotor parameters.

Parameter	Value	Parameter	Value
Γ_1	-0.9622	Γ_5	3.6441×10^{-4}
Γ_2	-0.0154	Γ_6	7.5395×10^{-5}
Γ_3	5.4716×10^{-5}	Γ_7	0.1308
Γ_4	1.0457	Γ_8	4.3753×10^{-5}

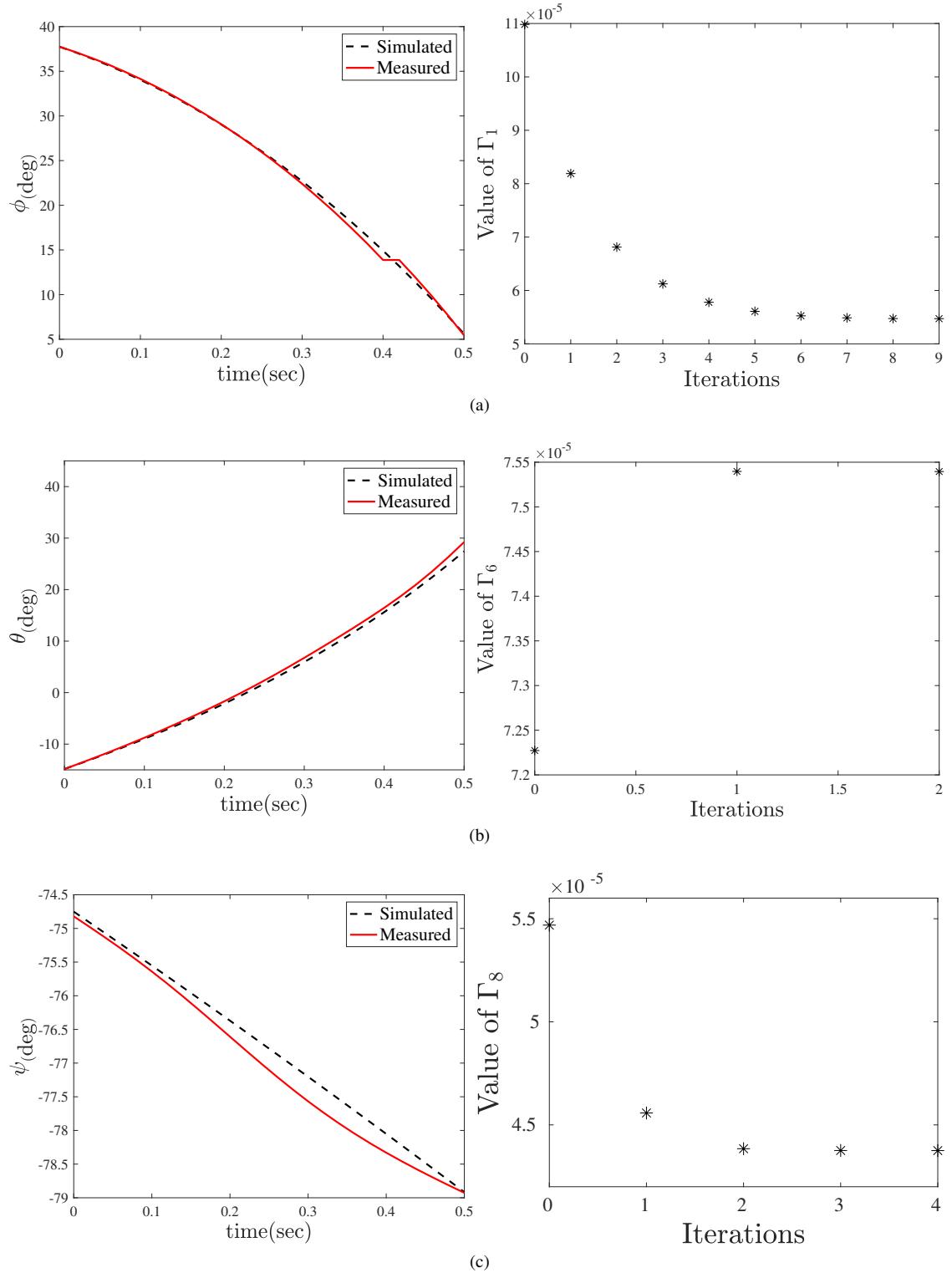


Figure 5: Identification process results when the quadrotor rotates about only one axis: (a) Identification of Γ_3 in free roll. (b) Identification of Γ_6 in free pitch. (c) Identification of Γ_8 in free yaw.

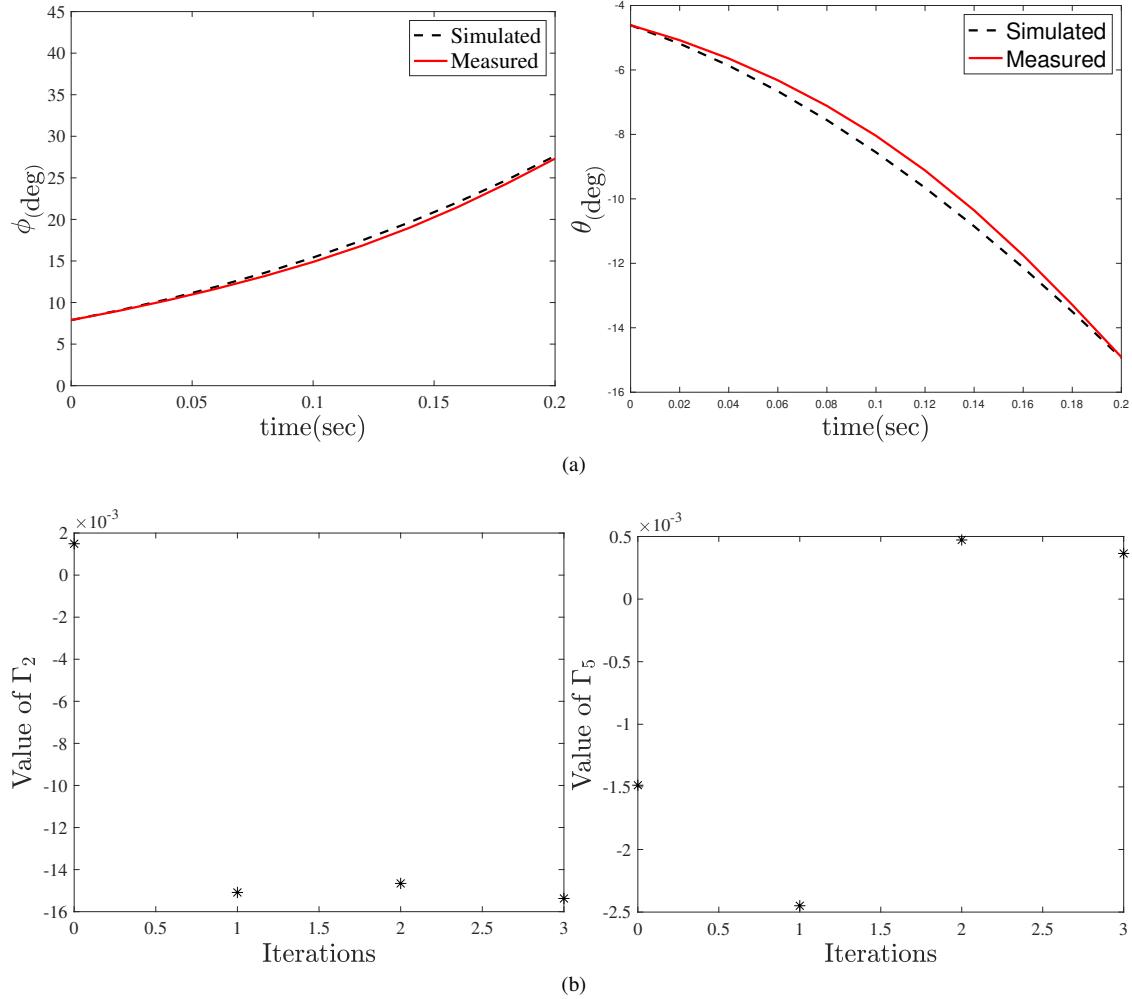


Figure 6: Identification process results when the quadrotor rotates about its roll and pitch axes: (a) Comparison of Simulation and experimental results. (b) Identification of Γ_2 and Γ_5 .

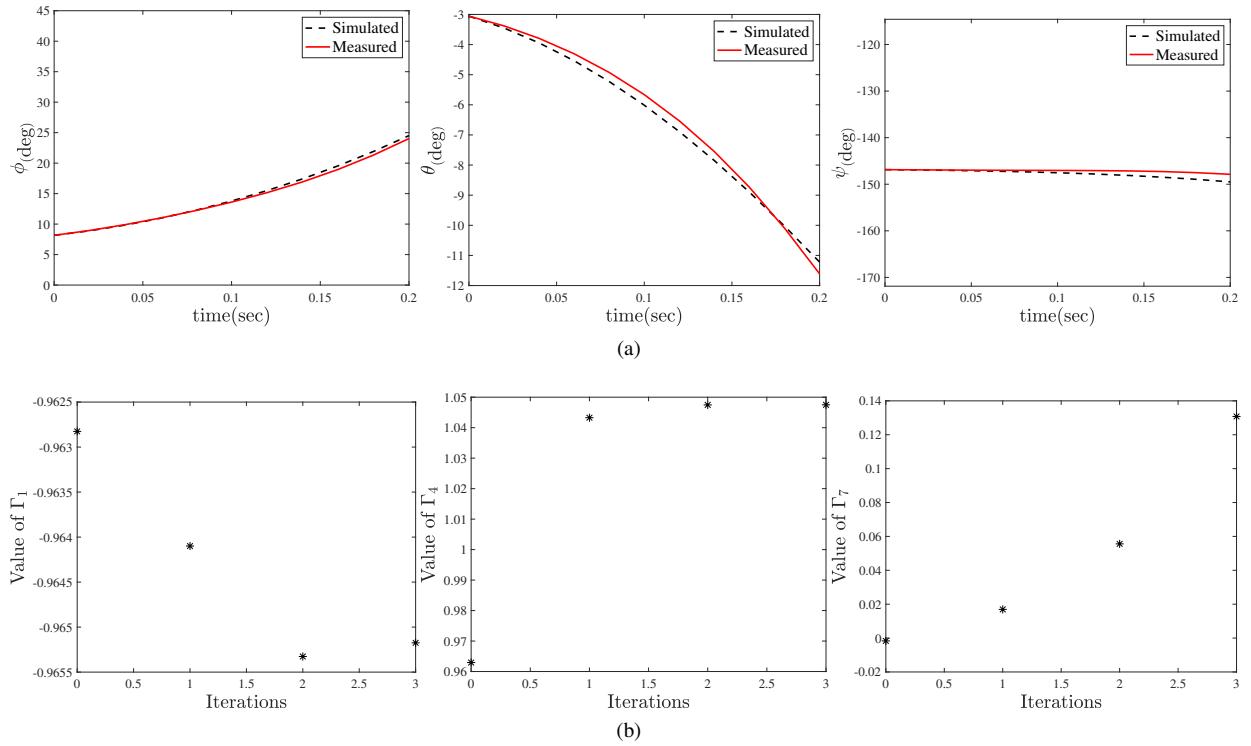


Figure 7: Identification process results when the quadrotor rotates about its roll, pitch, and yaw axes: (a) Comparison of Simulation and experimental results. (b) Identification of Γ_1 , Γ_4 and Γ_7 parameters.

5.2. Assessment of LQIR-DG Control Strategy

The performance of the LQIR-DG control strategy is assessed in three aspects: regulation and following of a square wave reference signal, disturbance rejection, and model uncertainty. The regulation and following of the square wave reference signal are examined in Section 5.2.1. The effectiveness of the controller in rejecting different levels of disturbance is evaluated in Section 5.2.2. Moreover, the impact of model uncertainty on the performance of the controller is studied in Section 5.2.3.

5.2.1. Regulating and Following Square Wave

This experiment assesses the efficacy of the LQIR-DG controller in regulating and following a square wave input for the quadrotor's attitude angles. The desired square wave input is generated for the roll and pitch angles, and the quadrotor's response to the input is measured. The results obtained demonstrate the robustness and precision of the LQIR-DG controller in accurately tracking the square wave input.

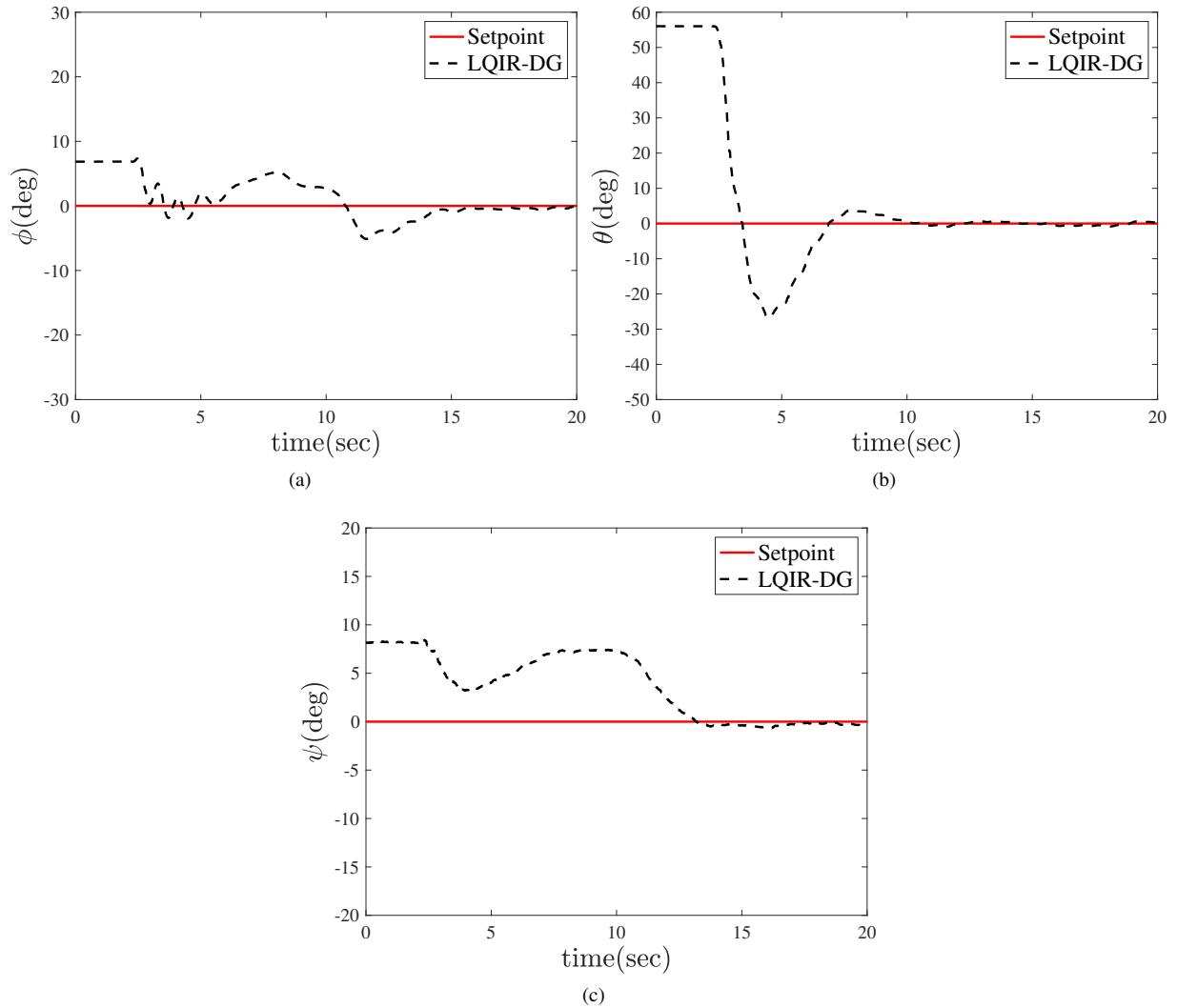


Figure 8: Efficiency Evaluation of the LQIR-DG Controller for (a) Roll, (b) Pitch, (c) and Yaw Attitude Angles

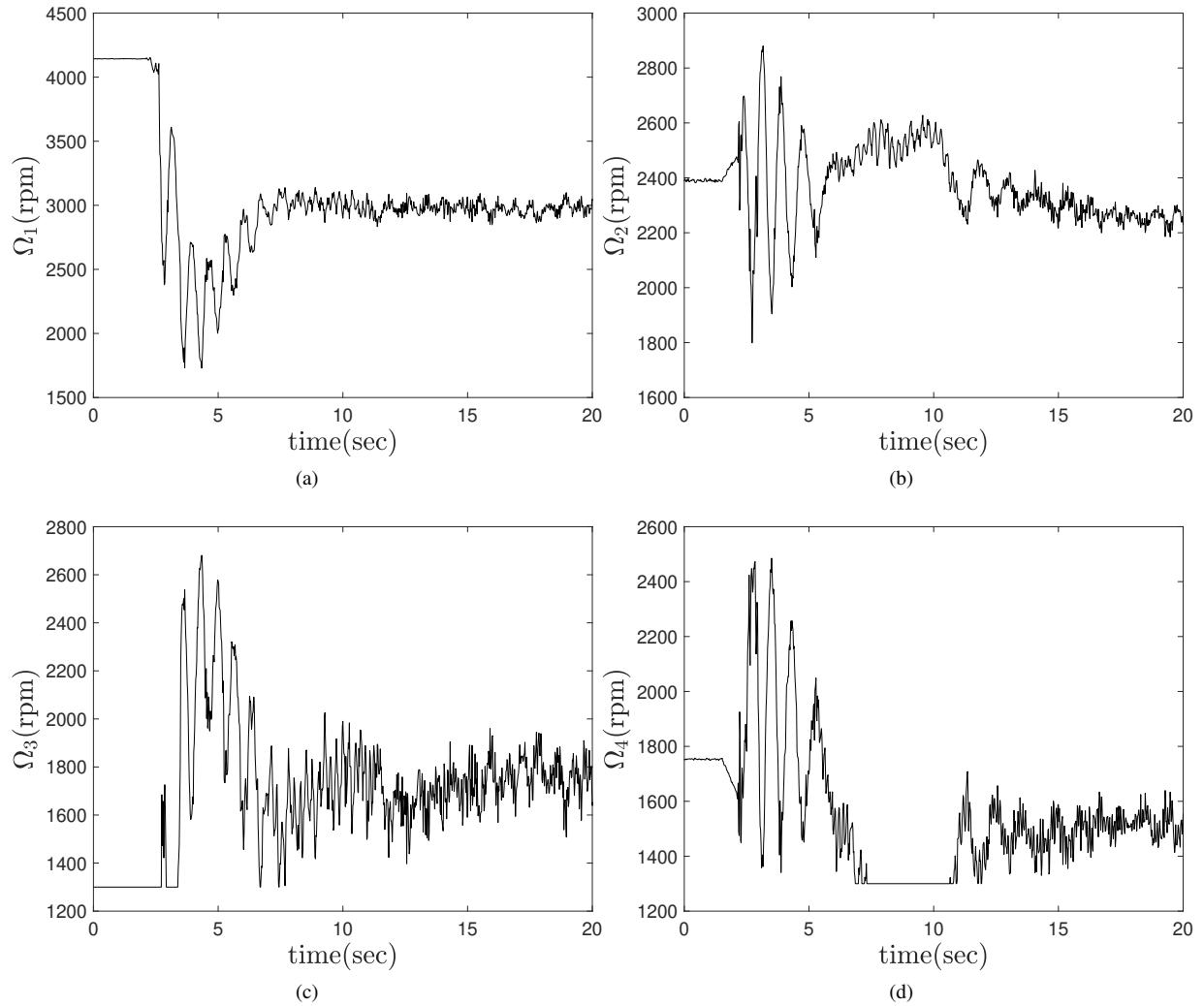


Figure 9: Temporal Evolution of Angular Velocity Commands for the LQIR-DG Controller

Figure 10 demonstrates the efficacy of the LQIR-DG controller in regulating the coupled roll and pitch channels to accurately track a desired square wave input with a frequency of 0.02 Hz and an amplitude of 20 degrees.

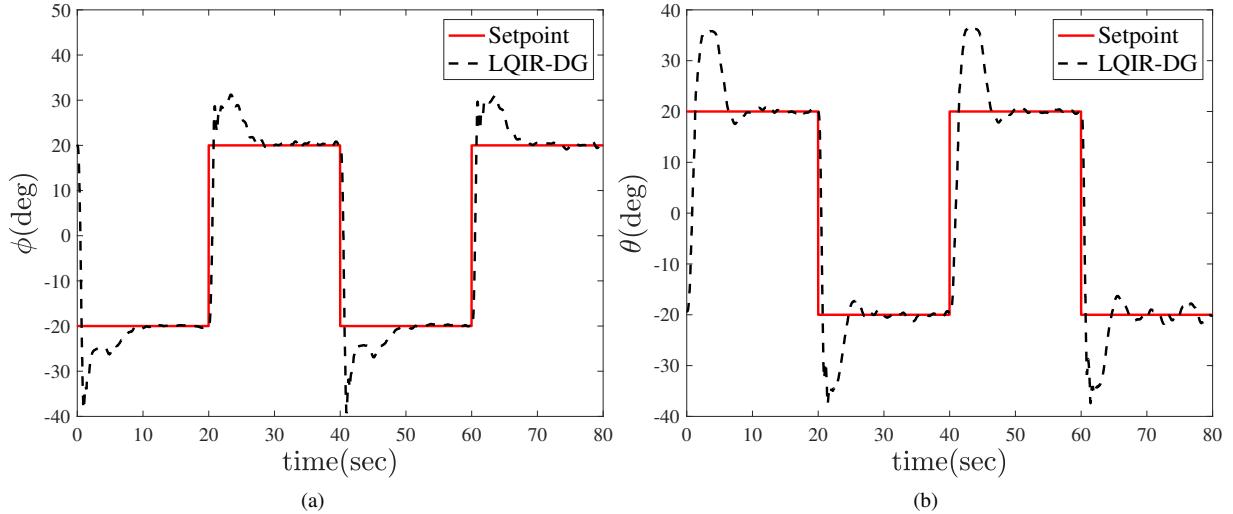


Figure 10: Performance evaluation of the LQIR-DG controller in tracking desired angles in the two-degree-of-freedom coupling mode. (a) Comparison of the actual and desired roll angles, and (b) comparison of the actual and desired pitch angles.

5.2.2. Investigating the possibility of disturbance rejection

This section investigates the capability of the LQIR-DG controller in rejecting input disturbances during regulation. To this end, a disturbance with an amplitude of 0.5 N is added to the input signal from 20 to 60 seconds. As depicted in Figure 11, the LQIR-DG controller demonstrates proficient coupling of the roll and screw channels, resulting in effective removal of the input disturbance. The performance of the controller is further evaluated by comparing the desired and actual roll angles in Figure 11(a), as well as the desired and actual pitch angles in Figure 11(b). The results indicate that the LQIR-DG controller performs appropriately in mitigating the input disturbance.

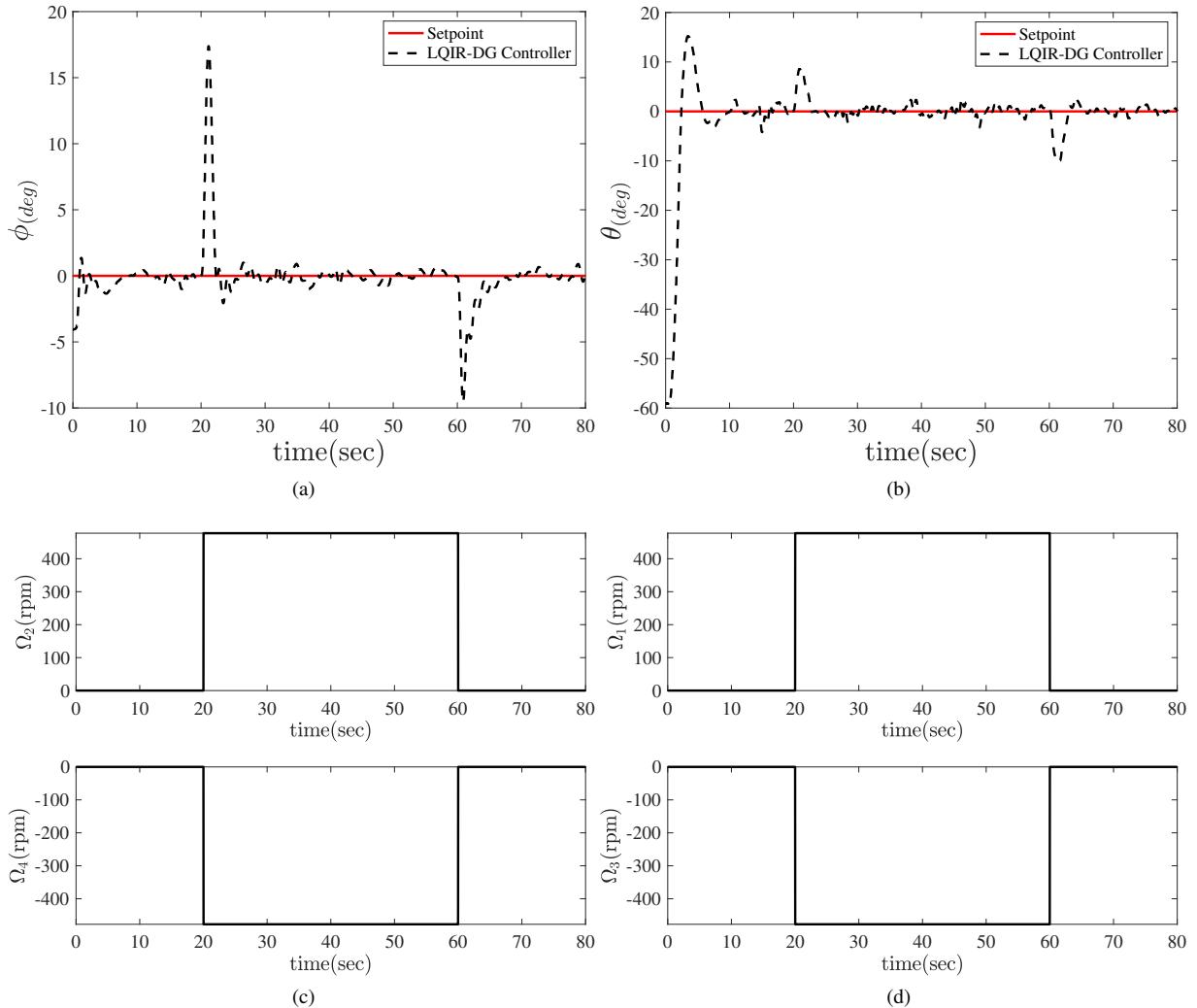


Figure 11: Evaluation of the LQIR-DG controller performance under the influence of input disturbance in the two-degree-of-freedom coupling mode. (a) illustrates the comparison between the desired roll angle and the actual value. Similarly, (b) shows the comparison between the desired pitch angle and the actual value.”

5.2.3. Investigating the impact of uncertainty in modeling

In this section, the performance of the LQIR-DG controller is investigated under the consideration of uncertainty in the 3DoF experimental setup modeling. Specifically, the controller’s effectiveness in the coupling mode of the roll and pitch channels is evaluated while taking into account the uncertain values of the moments of inertia around each axis of the body coordinate system. The results of this investigation are presented in figure 13, where a disturbance of 50 grams is added to the roll axis and 100 grams to the pitch axis.

Figure 13(a) displays the performance of the LQIR-DG controller in terms of the comparison between the desired roll angle and the actual roll angle, while figure 13(b) presents a similar comparison for the desired and actual pitch angles. The implementation results indicate that the LQIR-DG controller performs efficiently in achieving the desired values, even with the uncertainty in the moments of inertia.

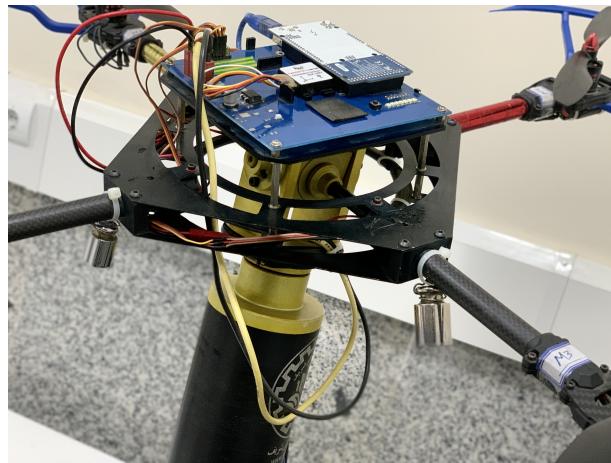


Figure 12: Quadrotor 3DoF Setup with Added Weight.

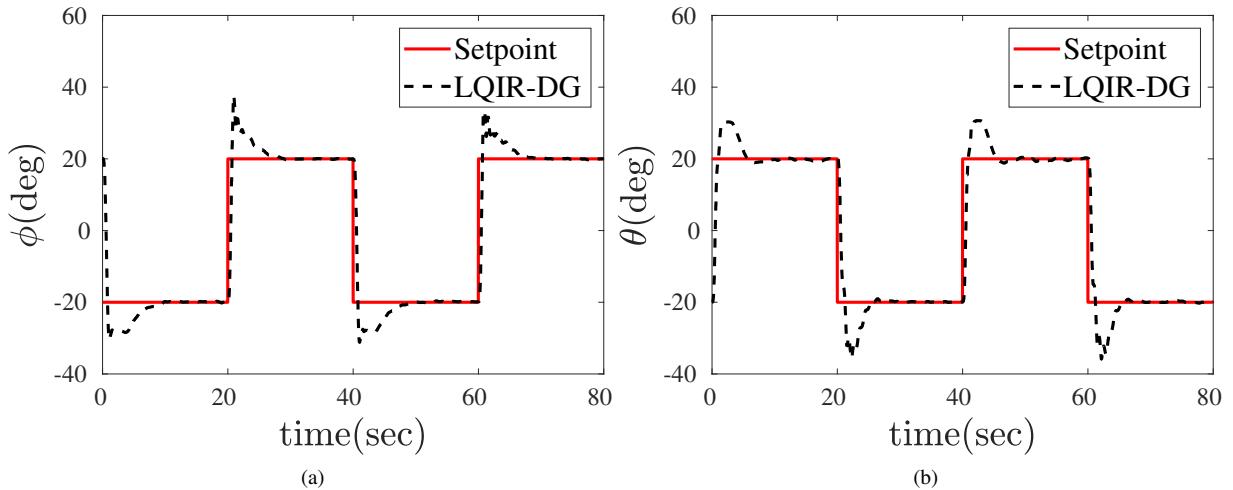


Figure 13: Assessing the performance of the LQIR-DG controller under uncertainty by adding weight to the roll and pitch axes in the two-degree-of-freedom coupling mode. Subfigure (a) shows the comparison between the desired and actual roll angle, while subfigure (b) displays the comparison between the desired and actual pitch angle.

5.2.4. Comparison with LQR, LQIR, and PID

This section compares the implementation performance of three different controllers for the quadrotor system: PID, LQR, and LQIR-DG. The comparison is made based on the implementation results of the controllers in the presence of uncertainty, input disturbance, and weight addition in the two-degree-of-freedom coupling mode.

The PID controller has been widely used in quadrotor control due to its simplicity and ease of implementation. However, its performance is limited in complex scenarios due to the inherent nonlinearity and uncertainty of the quadrotor system.

The LQR and LQIR controller is a linear feedback control method that can provide better performance than PID by optimizing a quadratic cost function. However, it requires an accurate model of the system, and is sensitive to uncertainties and disturbances in the system.

The LQIR-DG controller, which is based on game theory, considers both the system dynamics and the measurement uncertainties, and provides better performance in the presence of uncertainty and disturbances. In addition, the LQIR-DG controller can handle weight additions without requiring a re-design of the controller.

The implementation results indicate that the LQIR-DG controller outperforms the PID and LQR controllers in terms of accuracy and robustness in the presence of uncertainties and disturbances. The comparison results for the three controllers are summarized as follows:

- PID The PID controller can track the desired angles reasonably well in the absence of uncertainties and disturbances. However, its performance is significantly affected by the presence of uncertainty and disturbances, resulting in oscillations and overshoots in the response.
- LQR and LQIR The LQR and LQIR controller provides better performance than PID in the absence of uncertainties and disturbances. However, it is more sensitive to system uncertainties and disturbances, resulting in larger errors and oscillations in the response.
- LQIR-DG The LQIR-DG controller provides the best performance among the three controllers, as it can handle uncertainties, disturbances, and weight additions without requiring a re-design of the controller. The LQIR-DG controller provides accurate and robust control of the quadrotor system, with minimal errors and oscillations in the response.

In summary, the LQIR-DG controller provides better performance than PID and LQR controllers, and is a promising control method for quadrotor systems in the presence of uncertainties and disturbances.

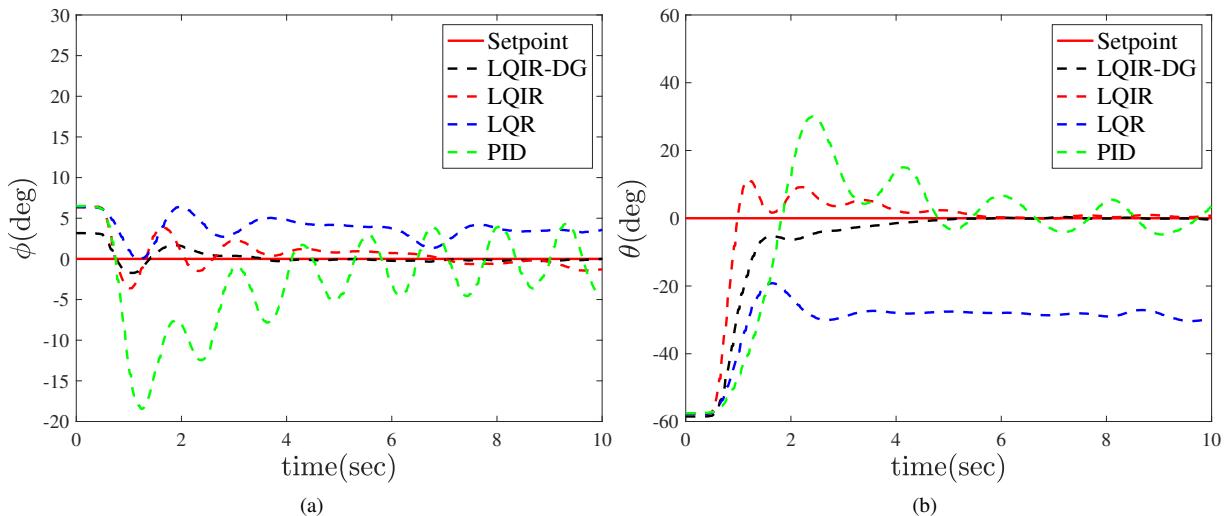


Figure 14: The implementation results of the LQIR-DG controller with added weight on the roll and pitch axes in the three-degree-of-freedom coupling mode. (a) Comparison of the desired roll angle with the actual roll angle, (b) Comparison of the desired pitch angle with the actual pitch angle.

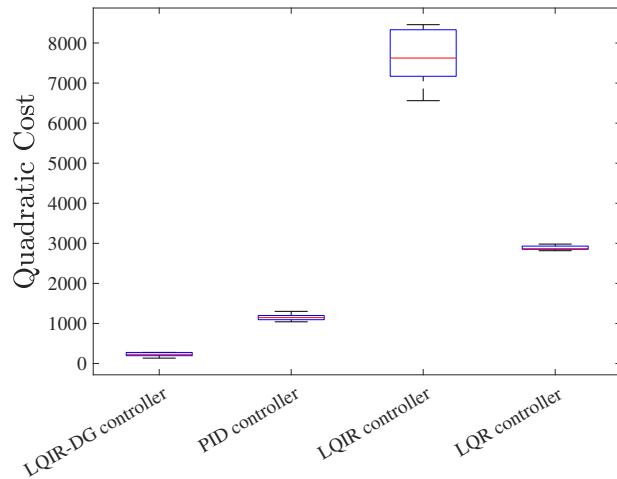


Figure 15: Comparison of the LQIR-DG controller with LQR, LQIR, and PID controllers using a quadratic cost function

6. Conclusion

This study has presented a novel approach for the implementation and evaluation of a linear quadratic with integral action controller based on differential game theory for level attitude control in an experimental quadrotor setup. The proposed controller design required an accurate model of the quadrotor in state-space form, which was linearized, and the model parameters were estimated. The LQIR-DG controller involved two players for each of the roll, pitch, and yaw channels. The first player optimized the control command for each channel based on a quadratic criterion, while the second player generated the worst disturbances.

The performance of the LQIR-DG controller was evaluated in level flight and compared to the LQR controller. The results showed the successful performance of the LQIR-DG method in level flight for attitude control, tracking square wave, disturbance rejection, and model uncertainty in the actual plant. Overall, this study has demonstrated the effectiveness and potential of the LQIR-DG method in quadrotor attitude control, which can be extended to other aerospace systems.

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