TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
20-010-01-01	MRB	05-55-25-200-802 05-55-25-200-803 05-55-25-200-804 05-55-25-200-805 05-55-25-200-806	1.1	15000 FH	15000 FH	ALL	ALL
	General visual inspection of HIRF/L sensitive wire runs outside the pressure vessel on left side of the airplane. Look for obvious signs of damage and lack of security of the wire runs.						
20-010-02-01	MRB	05-55-26-200-802 05-55-26-200-803 05-55-26-200-804 05-55-26-200-805 05-55-26-200-806	1.1	15000 FH	15000 FH	ALL	ALL
	General visual inspection of HIRF/L sensitive wire runs outside the pressure vessel on right side of the airplane. Look for obvious signs of damage and lack of security of the wire runs.						
20-020-00-01	MRB	05-55-15-200-806	1.1	30000 FH	30000 FH	ALL	ALL
	General visual inspection of HIRF/L sensitive wire runs inside the pressure vessel. Look for obvious signs of damage and lack of security of the wire runs.						
20-030-01-01	MRB	05-55-23-200-802 05-55-23-200-803 05-55-23-200-804 05-55-23-200-805 05-55-23-200-806	1.1	12000 FH	12000 FH	ALL	ALL
	Detail visual inspection of the Lightning/HIRF Protection components outside the pressure vessel on the left side of airplane. Inspect for condition of security and signs of corrosion.						
20-030-02-01	MRB	05-55-24-200-802 05-55-24-200-803 05-55-24-200-804 05-55-24-200-805 05-55-24-200-806	1.1	12000 FH	12000 FH	ALL	ALL
	Detail visual inspection of the Lightning/HIRF Protection components outside the pressure vessel on the right side of airplane. Inspect for condition of security and signs of corrosion.						
20-040-01-01	MRB	05-55-40-200-802 05-55-40-200-803 05-55-40-200-804 05-55-40-200-805 05-55-40-200-806 05-55-44-200-801 05-55-44-200-802 05-55-44-200-803 05-55-44-200-804 05-55-44-200-805	1.1	15000 FH	15000 FH	ALL	ALL
	Perform a functional check of the HIRF/L sensitive connectors outside the pressure vessel on the left side of the airplane. Check DC resistance from the backshell to ground. **AIRPLANE NOTE:** Functional check using the Loop Resistance Test in AMM 05-55-44-200-XXX is the Boeing preferred method. An alternate Bond Resistance Test is provided for operators in lieu of a Loop Resistance Test for operators choosing to utilize the approved Bond Resistance Test method. Please complete this functional check using a Loop Resistance Test or a Bond Resistance Test.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
20-040-02-01	MRB	05-55-41-200-802 05-55-41-200-803 05-55-41-200-804 05-55-41-200-805 05-55-41-200-806 05-55-45-200-801 05-55-45-200-802 05-55-45-200-803 05-55-45-200-804 05-55-45-200-805	1.1	15000 FH	15000 FH	ALL	ALL

Perform a functional check of the HIRF/L sensitive connectors outside the pressure vessel on the right side of the airplane. Check DC resistance from the backshell to ground.

AIRPLANE NOTE: Functional check using the Loop Resistance Test in AMM 05-55-45-200-XXX is the Boeing preferred method. An alternate Bond Resistance Test is provided for operators in lieu of a Loop Resistance Test for operators choosing to utilize the approved Bond Resistance Test method. Please complete this functional check using a Loop Resistance Test or a Bond Resistance Test.

20-040-04-01	MRB	05-55-42-200-805 05-55-46-200-803	1.1	15000 FH	15000 FH	ALL	ALL

Perform a functional check of the Lightning/HIRF protection components outside the pressure vessel in the vertical stabilizer. Check DC resistance from the backshell to ground.

AIRPLANE NOTE: Applicable to airplanes with the M2445 Rudder position sensor and M2446 Rudder actuator (servo) installed.

Functional check using the Loop Resistance Test in AMM 05-55-46-200-803 is the Boeing preferred method. An alternate Bond Resistance Test is provided for operators in lieu of a Loop Resistance Test for operators choosing to utilize the approved Bond Resistance Test method. Please complete this functional check using a Loop Resistance Test or a Bond Resistance Test.

20-040-05-01	MRB	05-55-44-200-806	1.1	2 YR	2 YR	ALL	ALL

Functionally check the Lightning/HIRF protection components (by performing a Loop Resistance Test) in the left wing to body fairing associated with disconnect bracket AC0520 for bond degradation.

AIRPLANE NOTE: Applicable to airplanes line number 1 thru 1856 that have not incorporated SB 737-24-1172.

20-040-05-02	MRB	05-55-45-200-806	1.1	2 YR	2 YR	ALL	ALL

Functionally check the Lightning/HIRF protection components (by performing a Loop Resistance Test) in the right wing to body fairing associated with disconnect bracket AD0520 for bond degradation.

AIRPLANE NOTE: Applicable to airplanes line number 1 thru 1856 that have not incorporated SB 737-24-1172.

20-040-06-01	MRB	05-55-44-200-806	1.1	16 YR	16 YR	ALL	ALL

Functionally check the Lightning/HIRF protection components (by performing a Loop Resistance Test) in the left wing to body fairing associated with disconnect bracket AC0520 for bond degradation.

AIRPLANE NOTE: Applicable to airplanes line number 1 thru 1856 that have incorporated SB 737-24-1172. Also applicable to airplanes line number 1857 and on.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
20-040-06-02	MRB	05-55-45-200-806	1.1	16 YR	16 YR	ALL	ALL
	Functionally check the Lightning/HIRF protection components (by performing a Loop Resistance Test) in the right wing to body fairing associated with disconnect bracket AD0520 for bond degradation. **AIRPLANE NOTE:** Applicable to airplanes line number 1 thru 1856 that have incorporated SB 737-24-1172. Also applicable to airplanes line number 1857 and on.						
20-050-00-01	MRB	05-55-15-200-804	1.1	20000 FH	20000 FH	ALL	ALL
	Perform a detail visual inspection of HIRF/L sensitive connectors inside the pressure vessel. During the inspection do not disconnect connectors. Look for condition, security, and signs of corrosion.						
20-060-00-01	MRB	05-55-43-200-801 05-55-43-200-802 05-55-43-200-803 05-55-43-200-804 05-55-43-200-805 05-55-43-200-807 05-55-43-200-808	1.1	30000 FH	30000 FH	ALL	ALL
	Functional check of HIRF/L sensitive connectors inside the pressure vessel by DC resistance check from backshell to ground.						
20-070-00-01	MRB	05-55-15-200-805	1.1	30000 FH	30000 FH	ALL	ALL
	Perform a detail visual inspection of the HIRF/L sensitive pig tails inside the pressure vessel look for condition of security and signs of corrosion.						
20-100-00-01	MRB	05-55-08-200-801 05-55-08-200-802 05-55-08-200-803 05-55-08-200-804 05-55-08-200-805 05-55-08-200-806 05-55-08-200-807 05-55-08-200-808	1.1	15000 FH	15000 FH	ALL	ALL
	Perform a general visual inspection of the bonding straps at the following locations: 1 strap on the rudder, 2 straps per aileron, leading edge flap, spoiler and each elevator surface. Strut to wing bonding uses 1 straps per wing. Each air conditioning pack compartment door and main landing gear door uses 2 bonding straps.						
20-110-01-01	MRB	05-55-10-200-801	1.1	6000 FH	6000 FH	ALL	ALL
	General visual inspection of external (cowl open) harness condition and security of left engine.						
20-110-02-01	MRB	05-55-10-200-801	1.1	6000 FH	6000 FH	ALL	ALL
	General visual inspection of external (cowl open) harness condition and security of right engine.						
20-120-01-01	MRB	05-55-10-220-801	1.1	15000 FH	15000 FH	ALL	ALL
	Detailed inspection of connectors for tightness (all connectors on harness J5, J6, J7, J8, J9, CJ9, J10, CJ10, MW0301, MW0302, MW0303 AND MW0304) on the left engine.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
20-120-02-01	MRB	05-55-10-220-801	1.1	15000 FH	15000 FH	ALL	ALL
	Detailed inspection of connectors for tightness (all connectors on harness J5, J6, J7, J8, J9, CJ9, J10, CJ10, MW0301, MW0302, MW0303 AND MW0304) on the right engine.						
20-130-01-01	MRB	05-55-11-200-801	1.1	15000 FH	15000 FH	ALL	ALL
	General visual inspection of the two engine bonding straps related to HIRF/Lightning for condition and security on left engine.						
20-130-02-01	MRB	05-55-11-200-801	1.1	15000 FH	15000 FH	ALL	ALL
	General visual inspection of the two engine bonding straps related to HIRF/Lightning for condition and security on right engine.						
20-140-00-01	MRB	05-55-42-200-802 05-55-42-200-804 05-55-46-200-801 05-55-46-200-802	1.1	15000 FH	15000 FH	ALL	ALL
	Perform a functional check of the HIRF/L sensitive connectors inside the empennage. NOTE: An alternate intrusive procedure (05-55-42-200-XXX) is provided for operators choosing to utilize the approved bonding meter method. **AIRPLANE NOTE:** Functional check using the Loop Resistance Test in AMM 05-55-46-200-803 is the Boeing preferred method. An alternate Bond Resistance Test is provided for operators in lieu of a Loop Resistance Test for operators choosing to utilize the approved Bond Resistance Test method. Please complete this functional check using a Loop Resistance Test or a Bond Resistance Test.						
20-141-00-01	MRB	05-55-46-200-804	1.1	48000 FH	48000 FH	ALL	ALL
	Functional check of the Lightning/HIRF protection components in the tail cone for electrical bond degradation using the Loop Resistance Tester (LRT). (L/HIRF) **AIRPLANE NOTE:** Applicable to airplanes line number 3470 and on or those airplanes that have incorporated SB 737-33-1146.						
20-142-01-01	MRB	05-55-47-200-801	1.1	18000 FH	18000 FH	ALL	ALL
	Functionally check the Lightning/HIRF protection components in the LRRA Antenna installation for degradation of the bond between antenna to structure using the Loop Resistance Tester (LRT). **AIRPLANE NOTE:** Applicable to airplanes with the S67-2002-18 LRRA antenna installed without Gasket AG723000-40. The production configuration for L/N 1 through 4306 is with LRRA antenna S67-2002-18 installed without Gasket AG723000-40.						
20-142-02-01	MRB	05-55-47-200-801	1.1	25000 FH	25000 FH	ALL	ALL
	Functionally check the Lightning/HIRF protection components in the LRRA Antenna installation for degradation of the bond between antenna to structure using the Loop Resistance Tester (LRT). **AIRPLANE NOTE:** Applicable to airplanes with the S67-2002-18 LRRA antenna and Gasket AG723000-40 installed. The production configuration for L/N 1 through 4306 is with LRRA antenna S67-2002-18 installed without Gasket AG723000-40.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
20-142-03-01	MRB	05-55-47-200-801	1.1	40000 FH	40000 FH	ALL	ALL
	Functionally check the Lightning/HIRF protection components in the LRRA Antenna installation for degradation of the bond between antenna to structure using the Loop Resistance Tester (LRT). **AIRPLANE NOTE:** Applicable to airplanes with the S67-2002-28 LRRA antenna and Gasket AG723000-40 installed. The production configuration for L/N 4307 and on is with LRRA antenna S67-2002-28 and Gasket AG723000-40 installed.						
20-290-00-01	MRB	05-42-01-100-801 20-60-02-100-801	1.1 1.2	36000 FC 12 YR	36000 FC 12 YR	ALL	ALL
	Restore (Clean) area forward of Nose Wheel Well. (EZAP) **INTERVAL NOTE:** Whichever comes first.						
20-300-00-01	MRB	05-42-01-100-803 20-60-02-100-801	1.1 1.2	36000 FC 12 YR	36000 FC 12 YR	ALL	ALL
	Restore (Clean) area above and outboard of Nose Wheel Well. (EZAP) **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Remove/displace insulation blankets as required.						
20-305-00-01	MRB	05-42-01-211-802 20-60-03-100-801	1.1 1.2	36000 FC 12 YR	36000 FC 12 YR	ALL	ALL
	Inspect (Detailed) the external power feeder wiring and connected EWIS in the area outboard of Nose Wheel Well (Right Side). (EZAP) **INTERVAL NOTE:** Whichever comes first.						
20-310-00-01	MRB	05-42-01-100-802 20-60-02-100-801	1.1 1.2	18000 FC 6 YR	18000 FC 6 YR	ALL	ALL
	Restore (Clean) areas behind the equipment racks in the Electrical and Electronics Compartment, and inside the Airstair Compartment (if installed). (EZAP) **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Access panel 117BL is only for airplanes with airstairs installed. Alternate access behind E2, E3 and E4 electronics racks is through 121JW, 121KW, 121LW and 122HW panels.						
20-320-00-01	MRB	05-42-01-211-801 20-60-03-100-801	1.1 1.2	18000 FC 6 YR	18000 FC 6 YR	ALL	ALL
	Inspect (Detailed) the IDG, APU starter/generator, battery, and external power feeder wiring and connected EWIS. (EZAP) **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Access through panels 121JW, 121KW, 121LW and 122HW is from Forward Cargo Compartment.						
20-325-00-01	MRB	05-42-01-210-803 20-60-04-100-801	1.1 1.2	18000 FC 6 YR	18000 FC 6 YR	ALL	ALL
	Inspect (General Visual) all exposed EWIS in the Electrical and Electronics Compartment excluding the IDG, APU starter/generator, battery, and external power feeder wiring and connected EWIS. (EZAP) **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Access through panels 121JW, 121KW, 121LW and 122HW is from Forward Cargo Compartment.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
20-330-00-01	MRB	05-42-01-100-807 20-60-02-100-801	1.1 1.2	18000 FC 6 YR	18000 FC 6 YR	ALL	ALL
	Restore (Clean) areas behind ceiling and sidewalls in the Forward Cargo Compartment. (EZAP)						
	INTERVAL NOTE: Whichever comes first.						
	ACCESS NOTE: Ceiling and sidewall panels removal required.						
20-335-00-01	MRB	05-42-01-210-802 20-60-04-100-801	1.1 1.2	18000 FC 6 YR	18000 FC 6 YR	ALL	ALL
	Inspect (General Visual) all exposed EWIS in the Forward Cargo Compartment. (EZAP)						
	INTERVAL NOTE: Whichever comes first.						
	ACCESS NOTE: Ceiling and sidewall panels removal required.						
20-340-00-01	MRB	05-42-01-100-806 20-60-02-100-801	1.1 1.2	18000 FC 6 YR	18000 FC 6 YR	ALL	ALL
	Restore (Clean) area below Forward Cargo Compartment. (EZAP)						
	INTERVAL NOTE: Whichever comes first.						
	ACCESS NOTE: Center floor panels removal required. Cargo loading system removed/displaced as required.						
20-350-00-01	MRB	05-42-01-100-805 20-60-02-100-801	1.1 1.2	36000 FC 12 YR	36000 FC 12 YR	ALL	ALL
	Restore (Clean) area aft of Forward Cargo Compartment. (EZAP)						
	INTERVAL NOTE: Whichever comes first.						
20-360-00-01	MRB	05-42-01-210-801 20-60-04-100-801	1.1 1.2	5500 FC 30 MO	5500 FC 30 MO	ALL	ALL
	Inspect (General Visual) all exposed EWIS in the Main Landing Gear Wheel Well. (EZAP)						
	INTERVAL NOTE: Whichever comes first.						
	ACCESS NOTE: Through main landing gear opening.						
20-370-00-01	MRB	05-42-01-100-804 20-60-02-100-801	1.1 1.2	18000 FC 6 YR	18000 FC 6 YR	ALL	ALL
	Restore (Clean) areas behind ceiling and sidewall panels in the Aft Cargo Compartment. (EZAP)						
	INTERVAL NOTE: Whichever comes first.						
	ACCESS NOTE: Ceiling and sidewall panels removal required.						
20-375-00-01	MRB	05-42-01-210-804 20-60-04-100-801	1.1 1.2	18000 FC 6 YR	18000 FC 6 YR	ALL	ALL
	Inspect (General Visual) all exposed EWIS in the Aft Cargo Compartment. (EZAP)						
	INTERVAL NOTE: Whichever comes first.						
	ACCESS NOTE: Ceiling and sidewall panels removal required.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
20-380-00-01	MRB	05-42-01-100-809 20-60-02-100-801	1.1 1.2	18000 FC 6 YR	18000 FC 6 YR	ALL	ALL
	Restore (Clean) area below Aft Cargo Compartment. (EZAP) **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Center floor panels removal required. Cargo loading system (if installed) removed/displaced as required.						
20-390-00-01	MRB	05-42-01-100-808 20-60-02-100-801	1.1 1.2	18000 FC 6 YR	18000 FC 6 YR	ALL	ALL
	Restore (Clean) area in Aft Cargo Compartment. (EZAP) **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Removal of aft cargo panels at Sta. 947 bulkhead required.						
20-400-00-01	MRB	05-42-02-100-806 20-60-02-100-801	1.1 1.2	18000 FC 6 YR	18000 FC 6 YR	ALL	ALL
	Restore (Clean) area in Flight Compartment from Sta. 186 to 211, WL 208 to 232. (EZAP) **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Accessible areas forward and above the rudder pedals.						
20-410-00-01	MRB	05-42-02-211-801 20-60-03-100-801	1.1 1.2	36000 FC 12 YR	36000 FC 12 YR	ALL	ALL
	Inspect (Detailed) all exposed EWIS in the P5 (fwd/aft), P6 and P18 panels. (EZAP) **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Opening P5 (fwd/aft), P6 and P18 panels required.						
20-415-00-01	MRB	05-42-02-210-801 20-60-04-100-801	1.1 1.2	36000 FC 12 YR	36000 FC 12 YR	ALL	ALL
	Inspect (General Visual) all exposed EWIS in the Flight Compartment excluding exposed EWIS in the P5, P6 and P18 panels. (EZAP) **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** With access provided. Seats removed. Control stand access panels, overhead & sidewall panels, glareshield, instruments/panels removal required.						
20-420-00-01	MRB	05-42-02-100-805 20-60-02-100-801	1.1 1.2	36000 FC 12 YR	36000 FC 12 YR	ALL	ALL
	Restore (Clean) areas above ceiling, behind sidewalls and under the raceway cover plates on both sides of the passenger compartment, aft of Flight Compartment to forward entry door. (EZAP) **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** With access provided. Galleys and lavs removed. Ceiling and sidewall panels removal required.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
20-430-00-01	MRB	05-42-02-100-804 20-60-02-100-801	1.1 1.2	36000 FC 12 YR	36000 FC 12 YR	ALL	ALL
	Restore (Clean) areas above ceiling, and behind sidewalls and return air grilles in Forward Passenger Compartment, Dry Area, Sta. 360 to 663.75. (EZAP) **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Removal of sidewall panels, ceiling panels and return air grilles required.						
20-435-00-01	MRB	05-42-02-100-803 20-60-02-100-801	1.1 1.2	36000 FC 8 YR	36000 FC 8 YR	ALL	ALL
	Restore (Clean) areas above ceiling, and behind sidewalls and return air grilles in Forward Passenger Compartment, Wet Areas, Sta. 360 to 663.75. (EZAP) **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Galleys and lavs removed. Removal of sidewall panels, ceiling panels and return air grilles required in areas where galleys and lavs are located.						
20-440-00-01	MRB	05-42-02-100-802 20-60-02-100-801	1.1 1.2	36000 FC 12 YR	36000 FC 12 YR	ALL	ALL
	Restore (Clean) areas above ceiling, and behind sidewalls and return air grilles in Aft Passenger Compartment, Dry Area, Sta. 663.75 to Aft Pressure Bulkhead. (EZAP) **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Removal of sidewall panels, ceiling panels and return air grilles required.						
20-445-00-01	MRB	05-42-02-100-801 20-60-02-100-801	1.1 1.2	36000 FC 8 YR	36000 FC 8 YR	ALL	ALL
	Restore (Clean) areas above ceiling, and behind sidewalls and return air grilles in Aft Passenger Compartment, Wet Areas, Sta. 663.75 to Aft Pressure Bulkhead. (EZAP) **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Galleys and lavs removed. Removal of sidewall panels, ceiling panels and return air grilles required in areas where galleys and lavs are located.						
20-450-00-01	MRB	05-42-03-211-802 20-60-03-100-801	1.1 1.2	36000 FC 12 YR	36000 FC 12 YR	ALL	ALL
	Inspect (Detailed) the APU starter/generator power feeder wiring and connected EWIS in the area aft of pressure bulkhead. (EZAP) **INTERVAL NOTE:** Whichever comes first.						
20-460-00-01	MRB	05-42-03-211-801 20-60-03-100-801	1.1 1.2	5500 FC 30 MO	5500 FC 30 MO	ALL	ALL
	Inspect (Detailed) the APU starter/generator power feeder wiring, ignition leads and connected EWIS in the APU Compartment. (EZAP) **INTERVAL NOTE:** Whichever comes first.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
20-465-00-01	MRB	05-42-03-210-801 20-60-04-100-801	1.1 1.2	18000 FC 6 YR	18000 FC 6 YR	ALL	ALL
	Inspect (General Visual) all exposed EWIS in the Tail Cone. (EZAP)						
	INTERVAL NOTE: Whichever comes first.						
20-470-00-01	MRB	05-42-04-211-801 20-60-03-100-801	1.1 1.2	5500 FC 30 MO	5500 FC 30 MO	ALL	ALL
	Inspect (Detailed) the IDG power feeder wiring and connected EWIS - Engine No. 1. (EZAP)						
	INTERVAL NOTE: Whichever comes first.						
20-480-00-01	MRB	05-42-04-211-802 20-60-03-100-801	1.1 1.2	5500 FC 30 MO	5500 FC 30 MO	ALL	ALL
	Inspect (Detailed) the IDG power feeder wiring and connected EWIS - Engine No. 2. (EZAP)						
	INTERVAL NOTE: Whichever comes first.						
20-490-00-01	MRB	05-42-04-211-803 20-60-03-100-801	1.1 1.2	18000 FC 6 YR	18000 FC 6 YR	ALL	ALL
	Inspect (Detailed) all exposed EWIS in the Forward Strut Fairing - Engine No. 1. (EZAP)						
	INTERVAL NOTE: Whichever comes first.						
20-500-00-01	MRB	05-42-04-211-804 20-60-03-100-801	1.1 1.2	18000 FC 6 YR	18000 FC 6 YR	ALL	ALL
	Inspect (Detailed) all exposed EWIS in Fan Cowl Support Beam - Engine No. 1. (EZAP)						
	INTERVAL NOTE: Whichever comes first.						
20-510-00-01	MRB	05-42-04-211-805 20-60-03-100-801	1.1 1.2	18000 FC 6 YR	18000 FC 6 YR	ALL	ALL
	Inspect (Detailed) all exposed EWIS in Strut Torque Box - Engine No. 1. (EZAP)						
	INTERVAL NOTE: Whichever comes first.						
20-520-00-01	MRB	05-42-04-211-806 20-60-03-100-801	1.1 1.2	18000 FC 6 YR	18000 FC 6 YR	ALL	ALL
	Inspect (Detailed) all exposed EWIS in the Forward Strut Fairing - Engine No. 2. (EZAP)						
	INTERVAL NOTE: Whichever comes first.						
20-530-00-01	MRB	05-42-04-211-807 20-60-03-100-801	1.1 1.2	18000 FC 6 YR	18000 FC 6 YR	ALL	ALL
	Inspect (Detailed) all exposed EWIS in Fan Cowl Support Beam - Engine No. 2. (EZAP)						
	INTERVAL NOTE: Whichever comes first.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
20-540-00-01	MRB	05-42-04-211-808 20-60-03-100-801	1.1 1.2	18000 FC 6 YR	18000 FC 6 YR	ALL	ALL
	Inspect (Detailed) all exposed EWIS in Strut Torque Box - Engine No. 2. (EZAP)						
	INTERVAL NOTE: Whichever comes first.						
20-550-00-01	MRB	05-42-05-211-801 20-60-03-100-801	1.1 1.2	18000 FC 6 YR	18000 FC 6 YR	ALL	ALL
	Inspect (Detailed) all exposed EWIS in the area from Leading Edge to Front Spar - Left Wing. (EZAP)						
	INTERVAL NOTE: Whichever comes first.						
	ACCESS NOTE: Leading edge flaps extended.						
20-560-00-01	MRB	05-42-05-211-802 20-60-03-100-801	1.1 1.2	36000 FC 12 YR	36000 FC 12 YR	ALL	ALL
	Inspect (Detailed) all exposed EWIS in the area from Leading Edge to Front Spar - Left Wing. (EZAP)						
	INTERVAL NOTE: Whichever comes first.						
	ACCESS NOTE: Access is gained by extending slats, and through lower wing surface access panels.						
20-570-00-01	MRB	05-42-05-211-803 20-60-03-100-801	1.1 1.2	18000 FC 6 YR	18000 FC 6 YR	ALL	ALL
	Inspect (Detailed) all exposed EWIS in the area from Rear Spar to Main Landing Gear Support Beam - Left Wing. (EZAP)						
	INTERVAL NOTE: Whichever comes first.						
20-580-00-01	MRB	05-42-05-211-804 20-60-03-100-801	1.1 1.2	18000 FC 6 YR	18000 FC 6 YR	ALL	ALL
	Inspect (Detailed) all exposed EWIS in the area from Rear Spar to Trailing Edge - Left Wing. (EZAP)						
	INTERVAL NOTE: Whichever comes first.						
	ACCESS NOTE: Flaps extended, spoilers raised.						
20-590-00-01	MRB	05-42-06-211-801 20-60-03-100-801	1.1 1.2	18000 FC 6 YR	18000 FC 6 YR	ALL	ALL
	Inspect (Detailed) all exposed EWIS in the area from Leading Edge to Front Spar - Right Wing. (EZAP)						
	INTERVAL NOTE: Whichever comes first.						
	ACCESS NOTE: Leading edge flaps extended.						
20-600-00-01	MRB	05-42-06-211-802 20-60-03-100-801	1.1 1.2	36000 FC 12 YR	36000 FC 12 YR	ALL	ALL
	Inspect (Detailed) all exposed EWIS in the area from Leading Edge to Front Spar - Right Wing. (EZAP)						
	INTERVAL NOTE: Whichever comes first.						
	ACCESS NOTE: Slats extended.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
20-610-00-01	MRB	05-42-06-211-803 20-60-03-100-801	1.1 1.2	18000 FC 6 YR	18000 FC 6 YR	ALL	ALL
	Inspect (Detailed) all exposed EWIS in the area from Rear Spar to Main Landing Gear Support Beam - Right Wing. (EZAP) **INTERVAL NOTE:** Whichever comes first.						
20-620-00-01	MRB	05-42-06-211-804 20-60-03-100-801	1.1 1.2	18000 FC 6 YR	18000 FC 6 YR	ALL	ALL
	Inspect (Detailed) all exposed EWIS in the area from Rear Spar to Trailing Edge - Right Wing. (EZAP) **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Flaps extended, spoilers raised.						
21-010-00-01	MRB	21-25-01-000-801 21-25-01-400-801	1.1	7500 FH	7500 FH	ALL	ALL
	Replace the recirculation fan HEPA filter(s). Note: Zone 126 is applicable to all 737NG aircraft. Zone 125 is applicable to 737-800/900 aircraft only. **ACCESS NOTE:** Access panel 122GW is applicable to all 737NG aircraft. Access panel 121EW is applicable to 737-800/900 aircraft only.						
21-015-00-01	MRB	21-25-03-200-801	1.1	6000 FH	6000 FH	ALL	ALL
	Perform a detailed inspection of the Recirculation Fan Check Valve(s) for condition, security, and proper operation.						
21-020-00-01	MRB	21-27-00-890-801	1.1	8000 FH	8000 FH	ALL	ALL
	Operationally check the alternate E/E cooling supply fan. Note: This task also checks the normal E/E cooling supply fan check valve.						
21-030-00-01	MRB	21-27-00-700-804	1.1	8000 FH	8000 FH	ALL	ALL
	Operationally check the alternate E/E cooling exhaust fan. Note: This task also checks the normal E/E cooling exhaust fan check valve.						
21-040-00-01	MRB	21-27-01-000-801 21-27-01-400-801	1.1	7500 FH	7500 FH	ALL	ALL
	Replace the E/E cooling supply fan filter.						
21-050-00-01	MRB	21-27-00-700-807	1.1	9000 FH	9000 FH	ALL	ALL
	Operationally check the equipment cooling overboard exhaust valve, supply fans, exhaust fans (if applicable), and recirculation fan(s) in smoke clearance mode. **AIRPLANE NOTE:** Exhaust fan operational check is applicable to 737-600/-700/-800 airplanes Line Number 1701 and on, and L/N 1-1700 that have incorporated SB 737-26-1122.						
21-060-00-01	MRB	21-31-00-710-801	1.1	12000 FH	12000 FH	ALL	ALL
	Operationally check the outflow valve manual mode (motor), selector panel, indicator, indicator feedback module.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
21-070-00-01	MRB	21-32-01-700-801 21-32-01-700-802	1.1	17000 FH	17000 FH	ALL	ALL
	Functionally check the positive pressure relief valves. (Note: The Boeing and the Hamilton Sundstrand test equipment are equivalent to each other, either test equipment can be used).						
21-080-00-01	MRB	21-32-02-000-801 21-32-02-400-801	1.1	20000 FH	20000 FH	ALL	ALL
	Replace the positive pressure relief valve filters.						
21-090-00-01	MRB	21-32-03-700-801	1.1	10 YR	10 YR	ALL	ALL
	Functionally check the negative pressure relief door.						
21-100-00-01	MRB	21-51-03-000-801	1.1	2000 FC	2000 FC	ALL	ALL
	Clean the primary and secondary heat exchangers.						
21-150-00-01	MRB	21-61-06-000-801 21-61-06-100-801 21-61-06-400-801	1.1	1200 FH	1200 FH	ALL	ALL
	Clean or replace the cabin temperature sensor filters (737-600/700 has one each in the control and passenger cabins, 737-800/900 has one in the control cabin and two in the passenger cabin).						
21-190-00-01	MRB	36-13-01-020-803 36-13-01-420-803	1.1	6 YR	6 YR	ALL	ALL
	Discard APU bleed air duct flexible pressure seal. **AIRPLANE NOTE:** Applicable to airplanes equipped with part number BOE2003-0052 seal.						
23-030-00-01	MRB	23-61-00-760-801	1.1	4 YR	4 YR	ALL	ALL
	Functional check of resistance of static discharges.						
23-040-00-01	MRB	23-71-00-710-801	1.1	300 FH	300 FH	ALL	ALL
	Operational check of the voice recorder and Recorder Independent Power Supply (RIPS) (if installed). **INTERVAL NOTE:** Or national requirement.						
23-050-00-01	MRB	23-71-00-730-801	1.1	4 YR	4 YR	ALL	ALL
	Functional check of the voice recorder for audio fidelity. **INTERVAL NOTE:** Or national requirement.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
23-060-00-01	MRB	23-71-21-000-801 23-71-21-400-801 23-71-21-700-801 23-71-21-700-802 23-71-21-700-803 23-71-21-700-804	1.1	NOTE		ALL	ALL
	Operational check of the ULB at battery replacement. **INTERVAL NOTE:** At battery replacement.						
23-070-00-01	MRB	23-71-21-000-801 23-71-21-400-801 23-71-21-960-801 23-71-21-960-802	1.1	VEN REC		ALL	ALL
	Replace ULB battery at vendor's recommendation. **INTERVAL NOTE:** At vendor's recommendation or national requirement.						
23-080-00-01	MRB	23-51-00-710-801	1.1	6000 FH	6000 FH	ALL	ALL
	Operational check of oxygen mask microphone.						
23-090-00-01	MRB	23-31-00-740-801	1.1	30000 FH	30000 FH	ALL	ALL
	Operational check of the passenger address speakers.						
23-100-00-02	MRB	23-24-00-730-802-002	1.1	NOTE		ALL	ALL
	Operationally check the Emergency Locator Transmitter (Automatic / Fixed Type). **AIRPLANE NOTE:** If Installed. **INTERVAL NOTE:** At Vendor's Recommendation.						
23-110-00-02	MRB	23-24-00-000-802-002 23-24-00-400-802-002	1.1	NOTE		ALL	ALL
	Discard the Emergency Locator Transmitter (Automatic / Fixed Type) Battery. **AIRPLANE NOTE:** If Installed. **INTERVAL NOTE:** At Vendor's Recommendation.						
24-010-01-01	MRB	12-13-21-600-802	1.1	1800 FH	1800 FH	ALL	ALL
	Change left IDG oil.						
24-010-02-01	MRB	12-13-21-600-802	1.1	1800 FH	1800 FH	ALL	ALL
	Change right IDG oil.						
24-020-01-01	MRB	12-13-21-200-802	1.1	800 FH	800 FH	ALL	ALL
	Detailed Inspection of left IDG delta P indicator.						
24-020-02-01	MRB	12-13-21-200-802	1.1	800 FH	800 FH	ALL	ALL
	Detailed Inspection of right IDG delta P indicator.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
24-030-01-01	MRB	12-13-21-200-801	1.1	800 FH	800 FH	ALL	ALL
	Detailed Inspection of left IDG oil level.						
24-030-02-01	MRB	12-13-21-200-801	1.1	800 FH	800 FH	ALL	ALL
	Detailed Inspection of right IDG oil level.						
24-040-01-01	MRB	24-11-41-000-801 24-11-41-200-801 24-11-41-400-801	1.1	1800 FH	1800 FH	ALL	ALL
	Replace left IDG charge and scavenge filters.						
24-040-02-01	MRB	24-11-41-000-801 24-11-41-200-801 24-11-41-400-801	1.1	1800 FH	1800 FH	ALL	ALL
	Replace right IDG charge and scavenge filters.						
24-050-01-01	MRB	24-11-61-200-801	1.1	3600 FH	3600 FH	ALL	ALL
	Torque check the left engine IDG quick attach/detach (QAD) coupling.						
24-050-02-01	MRB	24-11-61-200-801	1.1	3600 FH	3600 FH	ALL	ALL
	Torque check the right engine IDG quick attach/detach (QAD) coupling.						
24-100-00-01	MRB	24-34-00-710-802	1.1	15 DY	15 DY	ALL	ALL
	Operational check of the standby power control unit (SPCU). **AIRPLANE NOTE:** Applies to airplanes with dual battery installation only.						
24-110-00-01	MRB	24-31-41-710-801	1.1	15000 FC	15000 FC	ALL	ALL
	Check remote control circuit breaker. **AIRPLANE NOTE:** Applies to airplanes with dual battery installation only.						
24-120-00-01	MRB	24-31-11-000-802-002 24-31-11-400-802-002	1.1	1000 FH	1000 FH	ALL	ALL
	Restore the main and auxiliary batteries. **AIRPLANE NOTE:** Applies to airplanes with dual battery installations only. **INTERVAL NOTE:** Restore interval for 36 AMP/HR (small) battery is 1000 FH. 48 AMP/HR (large) is 2000 FH.						
24-130-00-01	MRB	24-41-11-200-802	1.1	5000 FC	5000 FC	ALL	ALL
	Functional check of the external power receptacle pins for excessive wear.						
24-140-00-01	MRB	24-41-11-200-804	1.1	5000 FC	5000 FC	ALL	ALL
	Detailed inspection of the external power receptacle pins for signs of overheat and security of installation.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
25-010-01-01	MRB	25-11-01-200-802	1.1	7500 FH	7500 FH	ALL	ALL
	Inspect (Detailed) the captains seat tracks and locking mechanism for wear, condition, and security.						
25-010-02-01	MRB	25-11-01-200-802	1.1	7500 FH	7500 FH	ALL	ALL
	Inspect (Detailed) the first officers seat tracks and locking mechanism for wear, condition, and security.						
25-020-00-01	MRB	25-11-00-200-802	1.1	600 FH	600 FH	ALL	ALL
	Inspect (Detailed) the captain, first officer, first observer, and second observer (if installed) seat harnesses, crotch straps, and shoulder belts (as applicable) for wear, condition, and security.						
25-030-00-01	MRB	25-11-00-200-801	1.1	3500 FH	3500 FH	ALL	ALL
	Operationally check the captain, first officer, and the first observer seat harness inertia reels.						
25-040-00-01	MRB	25-22-00-200-801	1.1	4000 FH	4000 FH	ALL	ALL
	Inspect (Detailed) the passenger seat belts w/o removal for wear, condition, and security.						
25-045-00-01	MRB	25-22-00-200-804	1.1	12000 FH	12000 FH	ALL	ALL
	Visually check the passenger compartment seats for proper attachment.						
25-050-00-01	MRB	25-22-00-710-802	1.1	12000 FH	12000 FH	ALL	ALL
	Operationally check, where applicable, the passenger seat break-over lock out feature on seats adjacent to emergency exits.						
25-070-00-01	MRB	25-22-00-710-803	1.1	6000 FH	6000 FH	ALL	ALL
	Functionally check, where applicable, the passenger seat back recline restriction mechanisms on seats adjacent to emergency exits.						
25-090-00-01	MRB	25-25-12-200-801	1.1	3000 FH	3000 FH	ALL	ALL
	Inspect (Detailed) the attendant seat harness and attachments without removal for wear, condition, and security.						
25-100-00-01	MRB	25-25-12-710-802	1.1	3000 FH	3000 FH	ALL	ALL
	Operationally check the attendant seat harness inertia reel lock feature.						
25-130-00-01	MRB	25-40-08-200-801	1.1	4000 FH	4000 FH	ALL	ALL
	Inspect (Detailed) the lavatory waste compartment flapper door and the waste compartment access door latching mechanism for wear, condition, and security. **INTERVAL NOTE:** FAA AD 74-08-09 interval is 1000 FH.						
25-160-00-01	MRB	25-52-00-200-803	1.1	450 FC	450 FC	ALL	ALL
	Inspect (General Visual) the fwd cargo compartment floor, ceiling, sidewall, bulkhead, and blowout (pressure relief) panels/liners for holes/tears, condition, and security.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
25-160-00-02	MRB	25-52-00-200-803	1.1	450 FC	450 FC	ALL	ALL
	Inspect (General Visual) the aft cargo compartment floor, ceiling, sidewall, bulkhead, and blowout (pressure relief) panels/liners for holes/tears, condition, and security.						
25-170-00-01	MRB	25-52-00-210-801	1.1	3000 FC	3000 FC	ALL	ALL
	Inspect (General Visual) the fwd cargo door restraint system for condition and security.						
25-170-00-02	MRB	25-52-00-210-801	1.1	3000 FC	3000 FC	ALL	ALL
	Inspect (General Visual) the aft cargo door restraint system for condition and security.						
25-190-01-01	MRB	25-61-10-710-801	1.1	10 YR	10 YR	ALL	ALL
	Inspect (Detailed) the left emergency exit hatch escape strap for condition and security.						
25-190-02-01	MRB	25-61-10-710-801	1.1	10 YR	10 YR	ALL	ALL
	Inspect (Detailed) the right emergency exit hatch escape strap for condition and security.						
25-210-00-01	MRB	25-61-10-210-801	1.1	10 YR	10 YR	ALL	ALL
	Inspect (Detailed) the flight compartment escape lanyards (2) for condition and security.						
25-220-00-01	MRB	25-66-00-710-801	1.1	NOTE		ALL	ALL
	Operational check of the entry and service door mounted emergency escape slide deployment system (on airplane). **INTERVAL NOTE:** Each operator is to perform an operational check of its Boeing 737NG entry and service door slide system to ensure its airline specific slide maintenance program is adequate. Each check from the operator's fleet shall include a minimum of one operational check of an installed slide from the left or right side at each door position, during each 6 year period. Checks shall alternate between the left and right door position. The total set would be 2 door minimum every 6 years.						
25-240-00-01	MRB	25-66-01-000-801 25-66-01-000-802 25-66-01-400-801 25-66-01-400-803	1.1	NOTE		ALL	ALL
	Restore the fwd entry emergency escape slide at the manufacturer's recommended interval. **INTERVAL NOTE:** Vendor Rec						
25-240-00-02	MRB	25-66-01-000-801 25-66-01-000-802 25-66-01-400-801 25-66-01-400-803	1.1	NOTE		ALL	ALL
	Restore the fwd galley emergency escape slide at the manufacturer's recommended interval. **INTERVAL NOTE:** Vendor Rec						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
25-240-00-03	MRB	25-66-01-000-801 25-66-01-000-802 25-66-01-400-801 25-66-01-400-803	1.1	NOTE		ALL	ALL
	Restore the aft entry emergency escape slide at the manufacturer's recommended interval. **INTERVAL NOTE:** Vendor Rec						
25-240-00-04	MRB	25-66-01-000-801 25-66-01-000-802 25-66-01-400-801 25-66-01-400-803	1.1	NOTE		ALL	ALL
	Restore the aft galley emergency escape slide at the manufacturer's recommended interval. **INTERVAL NOTE:** Vendor Rec						
25-290-00-01	MRB	25-64-00-900-801	1.1	VEN REC		ALL	ALL
	Restore the life jackets (if installed) at the manufacturer's recommended interval. **INTERVAL NOTE:** Vendor Rec						
25-330-00-01	MRB	25-64-00-710-801	1.1	2 YR	2 YR	ALL	ALL
	Operationally check the power megaphones.						
25-340-00-01	MRB	25-64-00-900-804	1.1	VEN REC		ALL	ALL
	Replace the power megaphone batteries at the manufacturer's recommended interval. **INTERVAL NOTE:** Vendor Rec						
25-350-00-01	MRB	25-64-00-000-801 25-64-00-400-801	1.1	VEN REC		ALL	ALL
	Functionally check (off-aircraft) the Emergency Locator Transmitter (Survival / Portable Type) per Vendor's CMM. **AIRPLANE NOTE:** If Installed. Applicable to dry cell type ELT's only. **INTERVAL NOTE:** At manufacturer's recommended interval or national regulatory requirement. Whichever comes first.						
25-360-00-01	MRB	25-64-00-000-801 25-64-00-400-801	1.1	VEN REC		ALL	ALL
	Discard the Emergency Locator Transmitter (Survival / Portable Type) batteries. **AIRPLANE NOTE:** If Installed. Applicable to non-dry cell type ELT's only. **INTERVAL NOTE:** At manufacturer's recommended interval or national regulatory requirement. Whichever comes first.						
25-370-00-01	MRB	25-64-00-210-802	1.1	2 YR	2 YR	ALL	ALL
	Visually check all detachable emergency equipment (gloves, smoke goggles, crash axe, flashlights, first aid kits, and medical kits, as applicable) for condition and presence.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
25-380-00-01	MRB	25-64-00-200-802	1.1	NOTE		ALL	ALL
	Operationally check the emergency flashlights. **INTERVAL NOTE:** At scheduled battery change or battery restoration.						
25-390-00-01	MRB	25-64-00-900-806	1.1	VEN REC		ALL	ALL
	Replace the emergency flashlight batteries at the manufacturer's recommended interval. **AIRPLANE NOTE:** Applicable to airplanes with non-rechargeable flashlight batteries.						
25-400-00-01	MRB	25-64-00-200-801	1.1	24 MO	24 MO	ALL	ALL
	Inspect (Detailed) the smoke hoods for condition.						
25-410-00-01	MRB	25-64-00-900-807	1.1	VEN REC		ALL	ALL
	Discard the smoke hoods at the manufacturer's recommended interval. **INTERVAL NOTE:** Vendor Rec						
25-420-00-01	MRB	25-64-00-900-808	1.1	VEN REC		ALL	ALL
	Restore the first aid kits at the manufacturer's recommended interval. **INTERVAL NOTE:** Vendor Rec						
25-430-00-01	MRB	25-64-00-900-809	1.1	VEN REC		ALL	ALL
	Restore the medical kits at the manufacturer's recommended interval. **INTERVAL NOTE:** Vendor Rec						
26-010-00-01	MRB	26-14-00-730-801	1.1	7500 FH	7500 FH	ALL	ALL
	Operational check of the lavatory smoke detectors.						
26-020-00-02	MRB	26-14-01-100-802	1.1	7500 FH	7500 FH	ALL	ALL
	Restore lavatory smoke detector grill by cleaning (Photoelectric type detectors).						
26-030-00-01	MRB	26-15-00-710-801	1.1	12000 FH	12000 FH	ALL	ALL
	Operational check of the APU remote fire detection system (M279 fire detection control module and P28 remote APU control panel).						
26-050-00-01	MRB	26-20-00-210-801 26-20-00-210-802	1.1	3600 FH	3600 FH	ALL	ALL
	Visually check engine fire bottle pressure gauge for correct pressure. (and APU fire bottle pressure gauge if installed).						
26-070-00-01	MRB	26-21-02-000-801 26-21-02-400-801	1.1	NOTE		ALL	ALL
	Replace the engine fire bottle squib cartridges. **INTERVAL NOTE:** AT VENDORS RECOMMENDATION						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
26-080-00-01	MRB	26-21-00-730-801	1.1	15000 FH	15000 FH	ALL	ALL
	Functional check of the engine squib firing circuit using the engine fire handle switch.						
26-090-00-01	MRB	26-21-01-000-801 26-21-01-400-801	1.1	NOTE		ALL	ALL
	Operational check of the engine fire extinguisher system check valves for freedom of movement. This operational check is accomplished at bottle discharge or at hydrostatic test, whichever comes first by moving check valve in a lateral motion while listening for movement of valve. **INTERVAL NOTE:** At bottle change.						
26-100-00-01	MRB	26-21-01-000-801 26-21-01-400-801	1.1	NOTE		ALL	ALL
	Functional check engine fire bottle pressure switch (off aircraft). **INTERVAL NOTE:** At vendor's recommendation.						
26-110-00-01	MRB	26-21-00-720-802	1.1	15000 FH	15000 FH	ALL	ALL
	Operational check of the engine fire handle (Including fire handle lock override circuitry) for engine shutdown and isolation.						
26-120-00-01	MRB	26-21-01-000-801 26-21-01-400-801	1.1	NOTE		ALL	ALL
	Functional check the engine fire bottle pressure gauge (off aircraft). **INTERVAL NOTE:** At vendor's recommendation.						
26-130-00-01	MRB	26-21-01-000-801 26-21-01-400-801	1.1	NOTE		ALL	ALL
	Replace the engine fire bottle over pressure relief disc. **INTERVAL NOTE:** At vendor's recommendation.						
26-150-00-01	MRB	26-21-00-720-801 26-21-00-730-803	1.1	15000 FH	15000 FH	ALL	ALL
	Functional check the engine fire extinguishing distribution system (flow and pressure check).						
26-170-00-01	MRB	26-21-01-210-801	1.1	15000 FH	15000 FH	ALL	ALL
	General visual inspection of engine fire extinguisher distribution system (including extinguisher bottles).						
26-210-00-01	MRB	26-22-02-000-801 26-22-02-400-801	1.1	NOTE		ALL	ALL
	Replace APU fire bottle squib. **INTERVAL NOTE:** At vendors recommendation.						
26-220-00-01	MRB	26-22-00-730-801	1.1	15000 FH	15000 FH	ALL	ALL
	Functional check of the APU squib firing circuit using the APU fire handle switch and verify squib circuit continuity.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
26-230-00-01	MRB	26-22-00-720-801	1.1	15000 FH	15000 FH	ALL	ALL
	Operational check of the APU fire handle (including fire handle lock override circuitry and P28 remote APU control panel) for APU shutdown and isolation.						
26-250-00-01	MRB	26-22-03-210-801	1.1	15000 FH	15000 FH	ALL	ALL
	General visual inspection of the APU remote control panel for condition and security of installation.						
26-260-00-01	MRB	26-22-01-000-801 26-22-01-400-801	1.1	NOTE		ALL	ALL
	Functional check the APU fire bottle pressure switch (off aircraft). **INTERVAL NOTE:** At vendor's recommendation.						
26-280-00-01	MRB	26-22-01-210-801	1.1	15000 FH	15000 FH	ALL	ALL
	Detail visual inspection of the APU fire extinguishing distribution system and APU fire bottle.						
26-290-00-01	MRB	26-24-01-900-801	1.1	NOTE		ALL	ALL
	Inspect (Detailed) lavatory waste compartment fire extinguishing bottles for correct weight. **INTERVAL NOTE:** At vendor recommendation.						
26-300-00-01	MRB	26-24-01-200-801	1.1	7500 FH	7500 FH	ALL	ALL
	Detail visual inspection of the lavatory fire bottle fusible tips and discharge tubes.						
26-330-00-01	MRB	26-23-02-000-801 26-23-02-400-801	1.1	LIF LIM		ALL	ALL
	Replace cargo fire bottle squib. **INTERVAL NOTE:** At vendors recommendation. **ACCESS NOTE:** Removal of air conditioning ducting may be required.						
26-340-00-01	MRB	26-23-00-720-801 26-23-00-730-803	1.1	12000 FH	12000 FH	ALL	ALL
	Functional check cargo fire extinguishing distribution system (flow and pressure check). **ACCESS NOTE:** Removal of air conditioning ducting may be required.						
26-360-00-01	MRB	26-23-01-000-801-001 26-23-01-400-802-001	1.1	NOTE		ALL	ALL
	Functional check the cargo fire bottle pressure switch (off aircraft). **INTERVAL NOTE:** At vendor's recommendation. **ACCESS NOTE:** Removal of air conditioning ducting may be required.						
26-370-00-01	MRB	26-23-00-730-801	1.1	7500 FH	7500 FH	ALL	ALL
	Operationally check the cargo fire extinguishing arm/discharge switches to verify circuitry. **ACCESS NOTE:** Removal of air conditioning ducting may be required.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
26-390-00-01	MRB	26-23-04-000-801 26-23-04-400-801	1.1	NOTE		ALL	ALL
	Replace the cargo compartment fire extinguishing system filter/drier. **AIRPLANE NOTE:** Filter/drier and metering orifice are applicable if second halon bottle installed. **INTERVAL NOTE:** At fire extinguisher bottle discharge. **ACCESS NOTE:** Removal of air conditioning ducting may be required.						
26-400-00-01	MRB	26-23-00-730-801	1.1	7500 FH	7500 FH	ALL	ALL
	Functionally check the cargo fire extinguishing timer. **AIRPLANE NOTE:** If second halon bottle installed. **ACCESS NOTE:** Removal of air conditioning ducting may be required.						
26-450-00-01	MRB	26-26-01-200-801	1.1	NOTE		ALL	ALL
	Inspect (Detailed) the portable halon fire extinguishers for proper pressure (if gauge installed), weight, and condition. **INTERVAL NOTE:** At vendors recommendation.						
26-470-00-01	MRB	26-26-02-200-801	1.1	1 YR	1 YR	ALL	ALL
	Inspect (detailed) the portable water fire extinguishers for condition.						
26-550-02-01	CMR	05-41-01-211-802	1.1 1.2	5500 FC 24 MO	5500 FC 24 MO	ALL	ALL
	Perform a detailed inspection of the Center Wing Rear Spar Vapor Web for cracking in the fiberglass panel between the vertical structural members. Do the inspection from inside the main gear wheel well. Note: This task card satisfies 26-CMR-02. **AIRPLANE NOTE:** Applicable to airplanes line number 2093, 2216 and on. **INTERVAL NOTE:** Whichever comes first.						
27-011-00-01	MRB	27-11-00-210-801	1.1	8000 FH	8000 FH	ALL	ALL
	General visual inspection of the forward aileron mechanical components.						
27-012-00-01	MRB	27-11-61-210-801	1.1	15000 FH	15000 FH	ALL	ALL
	Perform a general visual inspection of the aileron transfer mechanism.						
27-013-01-01	MRB	27-11-00-210-802	1.1	8000 FH	8000 FH	ALL	ALL
	Perform a general visual inspection of the left wing aileron mechanical components from the aileron PCU's to the aileron and the flight spoiler mechanical control path.						
27-013-02-01	MRB	27-11-00-210-802	1.1	8000 FH	8000 FH	ALL	ALL
	Perform a general visual inspection of the right wing aileron mechanical components from the aileron PCU's to the aileron and the flight spoiler mechanical control path.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
27-016-00-01	MRB	27-11-00-710-802	1.1	12000 FH	12000 FH	ALL	ALL
	Operationally check the aileron spring cartridge and transfer mechanism.						
27-018-00-01	MRB	27-11-00-700-806	1.1	25000 FH	25000 FH	ALL	ALL
	Functionally check the aileron spring cartridge and transfer mechanism.						
27-022-00-01	MRB	27-11-00-720-801	1.1	25000 FH	25000 FH	ALL	ALL
	Functionally check the force required to extend and collapse the A and B system aileron power control unit input pogo's.						
27-024-00-01	MRB	29-00-00-790-809	1.1	25000 FH	25000 FH	ALL	ALL
	Functionally check the A and B system aileron power control unit internal leakage in a loaded condition.						
27-026-01-01	MRB	12-22-11-600-801 12-22-11-640-801 12-22-11-640-802 12-22-11-640-803 12-22-11-640-804	1.1	4000 FH	4000 FH	ALL	ALL
	Lubricate the left wing aileron mechanical control path and aileron power control units. **SPECIAL NOTE:** CMR Task (27-CMR-11) interval for this task is 4,000 FH / 12 Months (whichever comes first) for airplanes using BMS 3-33 Grease and 3,000 FH / 9 Months (whichever comes first) for airplanes not using BMS 3-33 Grease. See MPD Section 9.						
27-026-02-01	MRB	12-22-11-600-801 12-22-11-640-801 12-22-11-640-802 12-22-11-640-803	1.1	4000 FH	4000 FH	ALL	ALL
	Lubricate the right wing aileron mechanical control path. **SPECIAL NOTE:** CMR Task (27-CMR-11) interval for this task is 4,000 FH / 12 Months (whichever comes first) for airplanes using BMS 3-33 Grease and 3,000 FH / 9 Months (whichever comes first) for airplanes not using BMS 3-33 Grease. See MPD Section 9.						
27-028-00-01	MRB	27-11-81-210-801	1.1	15000 FH	15000 FH	ALL	ALL
	Perform a detail visual inspection of the aileron feel and centering springs.						
27-030-00-01	MRB	27-11-00-700-807	1.1	15000 FH	15000 FH	ALL	ALL
	Operationally check, hydraulic power off, the aileron control surfaces for full range of travel and freedom of movement.						
27-032-00-01	MRB	27-11-00-700-803	1.1	25000 FH	25000 FH	ALL	ALL
	Functionally check the aileron system control wheel forces.						
27-033-00-01	MRB	27-09-91-200-806	1.1	8000 FH	8000 FH	ALL	ALL
	Functionally check the left wing aileron tab freeplay. **SPECIAL NOTE:** CMR task (27-CMR-12) interval for this task is 8,000 FH / 24 months, whichever comes first. See MPD Section 9.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
27-033-00-02	MRB	27-09-91-200-806	1.1	8000 FH	8000 FH	ALL	ALL
	Functionally check the right wing aileron tab freeplay. **SPECIAL NOTE:** CMR task (27-CMR-12) interval for this task is 8,000 FH / 24 months, whichever comes first. See MPD Section 9.						
27-034-01-01	MRB	27-11-31-210-801	1.1 1.2	8000 FH 3 YR	8000 FH 3 YR	ALL	ALL
	Perform a detail visual inspection of the left wing aileron balance bay seals. **INTERVAL NOTE:** Whichever occurs first.						
27-034-02-01	MRB	27-11-31-210-801	1.1 1.2	8000 FH 3 YR	8000 FH 3 YR	ALL	ALL
	Perform a detail visual inspection of the right wing aileron balance bay seals. **INTERVAL NOTE:** Whichever occurs first.						
27-035-00-01	MRB	27-09-91-200-801	1.1	16000 FH	16000 FH	ALL	ALL
	Functionally check the left wing aileron surface freeplay.						
27-035-00-02	MRB	27-09-91-200-801	1.1	16000 FH	16000 FH	ALL	ALL
	Functionally check the right wing aileron Surface freeplay.						
27-036-00-01	MRB	27-21-61-210-801	1.1	12000 FH	12000 FH	ALL	ALL
	Detail Inspection of Aft Rudder Quadrant, Torque Tube Assembly, Feel and Centering Unit, and associated Input/Output Rods.						
27-038-00-01	MRB	27-21-51-210-801	1.1	15000 FH	15000 FH	ALL	ALL
	General visual inspection of the rudder forward mechanical control path.						
27-040-00-01	MRB	12-22-21-600-802	1.1	6000 FH	6000 FH	ALL	ALL
	Lubricate the rudder feel and centering unit spring slider.						
27-041-00-01	MRB	27-21-00-700-824-002	1.1	10000 FH	10000 FH	ALL	ALL
	Operationally check the main rudder PCU force fight monitor. **AIRPLANE NOTE:** Applicable to airplanes line number 596; and 1268 and on; and L/N 1-595 and 597-1267 with incorporation of SB 737-27-1253.						
27-043-00-01	MRB	27-21-00-700-823-002	1.1	12500 FH	12500 FH	ALL	ALL
	Operationally check the rudder PCU overrides. **AIRPLANE NOTE:** Applicable to airplanes line number 596; and 1268 and on; and L/N 1-595 and 597-1267 with incorporation of SB 737-27-1253.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
27-046-00-01	MRB	12-22-21-600-801 12-22-21-640-801	1.1	6000 FH	6000 FH	ALL	ALL
	Lubricate the main and standby rudder power control unit rod ends and rudder hinges.						
27-047-00-01	MRB	27-21-00-700-822-002	1.1	10000 FH	10000 FH	ALL	ALL
	Functionally check the rudder power control unit internal leakage in a loaded condition. **SPECIAL NOTE:** CMR Task (27-CMR-10) interval for this task is 10,000 FH. See MPD Section 9. **AIRPLANE NOTE:** Applicable to airplanes line number 596; and 1268 and on; and L/N 1-595 and 597-1267 with incorporation of SB 737-27-1253. **INTERVAL NOTE:** MSG-3 analysis for this task is 12500 FH. CMR interval for this task is 10000 FH. See MPD Section 9.						
27-048-00-01	MRB	27-21-00-700-822-002	1.1	10000 FH	10000 FH	ALL	ALL
	Functionally check the rudder PCU relief valve (Authority Limiter) during rudder PCU internal leakage. Note: This task is satisfied by accomplishment of task 27-047-00. **AIRPLANE NOTE:** Applicable to airplanes line number 596; and 1268 and on; and L/N 1-595 and 597-1267 with incorporation of SB 737-27-1253.						
27-054-00-01	MRB	29-00-00-790-808	1.1	25000 FH	25000 FH	ALL	ALL
	Functionally check the standby rudder power control unit for internal leakage in a loaded condition.						
27-056-00-01	MRB	27-21-24-210-801	1.1	5000 FH	5000 FH	ALL	ALL
	Perform a general visual inspection of the standby rudder power control unit with hydraulic power on.						
27-058-00-01	MRB	27-21-24-210-802	1.1	15000 FH	15000 FH	ALL	ALL
	Perform a detail visual inspection of the standby rudder power control unit fore and aft attachment points (structure to PCU to rudder surface).						
27-060-00-02	MRB	27-21-00-700-820-002	1.1	15000 FH	15000 FH	ALL	ALL
	Operationally check the wheel to rudder interconnect system (WTRIS).						
27-062-00-02	MRB	27-21-00-700-819-002	1.1	15000 FH	15000 FH	ALL	ALL
	Operationally check the rudder trim through the full range of movement.						
27-064-00-02	MRB	27-21-00-700-814-002	1.1	16000 FH	16000 FH	ALL	ALL
	Functionally check the rudder surface freeplay.						
27-068-00-01	MRB	27-31-20-740-801	1.1	15000 FH	15000 FH	ALL	ALL
	Operationally check the elevator feel shift function. **AIRPLANE NOTE:** If Installed						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
27-070-00-01	MRB	12-22-31-600-801 12-22-31-600-802 12-22-31-640-801 12-22-31-640-802	1.1	6000 FH	6000 FH	ALL	ALL
	Lubricate the left elevator mechanical control path.						
27-070-00-02	MRB	12-22-31-600-801 12-22-31-600-802 12-22-31-640-801 12-22-31-640-802	1.1	6000 FH	6000 FH	ALL	ALL
	Lubricate the right elevator mechanical control path.						
27-073-00-01	MRB	27-31-61-210-801	1.1	15000 FH	15000 FH	ALL	ALL
	Perform a general visual inspection of the elevator mechanical control path.						
27-074-00-01	MRB	27-31-00-700-807	1.1	15000 FH	15000 FH	ALL	ALL
	Operationally check, hydraulic power off, the elevator control surfaces for full range of travel and freedom of movement.						
27-075-01-01	MRB	27-31-34-210-801	1.1	7500 FH	7500 FH	ALL	ALL
	Perform a general visual inspection of the left elevator balance weight installation and elevator tab control mechanism.						
27-075-02-01	MRB	27-31-34-210-801	1.1	7500 FH	7500 FH	ALL	ALL
	Perform a general visual inspection of the right elevator balance weight installation and elevator tab control mechanism.						
27-076-00-01	MRB	27-31-14-210-801	1.1	12000 FH	12000 FH	ALL	ALL
	Perform a general visual inspection of the elevator power control units with hydraulic power on.						
27-078-00-01	MRB	29-00-00-790-809	1.1	25000 FH	25000 FH	ALL	ALL
	Functionally check the A and B system elevator power control unit for internal leakage in a loaded condition.						
27-080-00-01	MRB	27-31-17-200-801 27-31-17-790-801	1.1	14000 FH	14000 FH	ALL	ALL
	Drain and leak check the elevator pitot-static system.						
27-084-00-01	MRB	27-31-00-720-801	1.1	25000 FH	25000 FH	ALL	ALL
	Functionally check the force necessary to breakout the elevator control column override assembly.						
27-086-00-01	MRB	27-31-00-700-815	1.1	6000 FH	6000 FH	ALL	ALL
	Operationally check the elevator tab control system.						
27-088-00-01	MRB	27-32-00-710-801 27-32-00-740-803	1.1	15000 FH	15000 FH	ALL	ALL
	Functionally check the stall warning system.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
27-092-00-01	MRB	27-31-00-820-809	1.1	25000 FH	25000 FH	ALL	ALL
	Functionally check the force necessary to collapse and extend the elevator input rod pogo's.						
27-093-00-01	MRB	27-31-00-220-802	1.1 1.2	2000 FC 4000 FH	2000 FC 4000 FH	ALL	ALL
	Perform a detailed visual inspection of the left elevator tab and left elevator tab mechanism. **SPECIAL NOTE:** CMR Task (27-CMR-07) interval for this task is 2,000 CYC / 4,000 FH, whichever comes first. See MPD Section 9. **AIRPLANE NOTE:** Applicable to airplanes line number 596; and 1175 and on; and L/N 1-595 and 597-1174 that have incorporated SB 737-55A1080. **INTERVAL NOTE:** Whichever comes first.						
27-093-00-02	MRB	27-31-00-220-802	1.1 1.2	2000 FC 4000 FH	2000 FC 4000 FH	ALL	ALL
	Perform a detailed visual inspection of the right elevator tab and right elevator tab mechanism. **SPECIAL NOTE:** CMR Task (27-CMR-07) interval for this task is 2,000 CYC / 4,000 FH, whichever comes first. See MPD Section 9. **AIRPLANE NOTE:** Applicable to airplanes line number 596; and 1175 and on; and L/N 1-595 and 597-1174 that have incorporated SB 737-55A1080. **INTERVAL NOTE:** Whichever comes first.						
27-094-00-01	MRB	22-11-26-710-801	1.1	15000 FH	15000 FH	ALL	ALL
	Perform a BITE check of the elevator autopilot servo pressure regulator.						
27-098-01-01	MRB	27-31-32-200-804	1.1	8000 FH	8000 FH	ALL	ALL
	Functionally check the left and right elevator surface freeplay.						
27-099-00-01	MRB	27-31-32-200-805	1.1 1.2	2000 FC 4000 FH	2000 FC 4000 FH	ALL	ALL
	Functionally check the elevator tab freeplay. **SPECIAL NOTE:** CMR Task (27-CMR-08) interval for this task is 2,000 CYC / 4,000 FH, whichever comes first. See MPD Section 9. **AIRPLANE NOTE:** Applicable to all airplanes except 737-900 line number 683 to 1174 that have not incorporated SB 737-55-1081. **INTERVAL NOTE:** Whichever comes first.						
27-099-00-02	MRB	27-31-32-200-805	1.1 1.2	2000 FC 4000 FH	2000 FC 4000 FH	ALL	ALL
	Functionally check the elevator tab freeplay. **SPECIAL NOTE:** CMR Task (27-CMR-08) interval for this task is 2,000 CYC / 4,000 FH, whichever comes first. See MPD Section 9. **AIRPLANE NOTE:** Applicable to all airplanes except 737-900 line number 683 to 1174 that have not incorporated SB 737-55-1081. **INTERVAL NOTE:** Whichever comes first.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
27-100-00-01	MRB	27-31-00-200-801	1.1	6 YR	6 YR	ALL	ALL
	Perform a detailed visual inspection of the elevator push rods (between output torque tube and control surface) and attachment bolts.						
27-101-00-01	MRB	27-31-00-200-802	1.1	6 YR	6 YR	ALL	ALL
	Perform a detailed visual inspection of the single element dual load path feel and centering unit output rod.						
27-102-00-01	MRB	12-22-41-600-801	1.1 1.2	1600 FH 1 YR	1600 FH 1 YR	ALL	ALL
	Lubricate the stabilizer trim actuator and actuator gimbal pins and ballnut. **INTERVAL NOTE:** Whichever comes first.						
27-104-00-01	MRB	12-22-41-600-802	1.1	25000 FH	25000 FH	ALL	ALL
	Lubricate the forward stabilizer trim mechanism drive train.						
27-106-00-01	MRB	27-41-00-700-808	1.1	15000 FH	15000 FH	ALL	ALL
	Functionally check the secondary stabilizer trim brake.						
27-107-00-01	MRB	12-22-41-610-802	1.1	7500 FH	7500 FH	ALL	ALL
	Service the HORIZONTAL TRIM ACTUATOR BRAKE. **AIRPLANE NOTE:** Applicable to airplanes line number 350 and on. Applicable to airplanes line number 1 to 349 that have incorporated SB 737-27-1210.						
27-108-00-01	MRB	27-41-81-000-801 27-41-81-400-801	1.1	25000 FH	25000 FH	ALL	ALL
	Remove the stabilizer trim actuator for restoration. **SPECIAL NOTE:** CMR Task (27-CMR-02) interval for this task is 4,500 FH. See MPD Section 9. **AIRPLANE NOTE:** For 251A4510-4 actuator, restore both primary and secondary stabilizer trim actuator brakes. For 251A4510-5 actuator, restore only the secondary stabilizer trim actuator brake. **INTERVAL NOTE:** For 251A4510-6,-9, -10 and on actuators, interval is 25000FH. CMR interval for 251A4510-4 and -5 actuators is 4500 FH.						
27-110-00-01	MRB	27-41-81-210-801	1.1 1.2	6400 FH 2 YR	6400 FH 2 YR	ALL	ALL
	Perform detail visual inspection of the stabilizer trim jackscrew, ballnut, ballnut return tubes, and the upper and lower gimbal pins. **INTERVAL NOTE:** Whichever comes first.						
27-112-00-01	MRB	27-41-00-700-805	1.1	15000 FH	15000 FH	ALL	ALL
	Operationally check the stabilizer trim limit switches (flaps up and flaps down).						
27-114-00-01	MRB	27-41-00-710-801	1.1	6000 FH	6000 FH	ALL	ALL
	Operationally check the stabilizer trim control column switching module.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
27-116-00-01	MRB	27-41-91-700-801	1.1	12000 FH	12000 FH	ALL	ALL
	Operationally check the Main Electric Horizontal Stabilizer Trim Cutout Switch (Control Stand).						
27-118-00-01	MRB	27-41-00-700-807	1.1	15000 FH	15000 FH	ALL	ALL
	Operationally check the aisle stand stabilizer trim override switch.						
27-120-00-01	MRB	27-41-41-210-801	1.1	15000 FH	15000 FH	ALL	ALL
	Detail visual inspection of the upper and lower stabilizer trim forward mechanism retention turnbuckles and turnbuckle attachment points.						
27-121-00-01	MRB	27-41-41-210-803 27-62-00-210-801	1.1	15000 FH	15000 FH	ALL	ALL
	Perform a general visual inspection of the forward stabilizer trim mechanism and speedbrake lever assembly and auto speedbrake electric actuator, actuator rod end, and forward attachment point.						
27-122-00-01	MRB	27-41-97-700-801	1.1	15000 FH	15000 FH	ALL	ALL
	Operationally check the control column stabilizer trim arm and directional switches for movement of a single switch to cause stabilizer movement.						
27-132-00-01	MRB	12-22-51-610-801	1.1	5000 FC	5000 FC	ALL	ALL
	Check flap power drive unit oil level and service as required.						
27-134-00-01	MRB	12-22-51-610-802	1.1	25000 FC	25000 FC	ALL	ALL
	Replace the flap power drive unit oil.						
27-136-01-01	MRB	12-22-51-640-807 12-22-51-640-808 12-22-51-640-809 12-22-51-640-810	1.1 1.2	4000 FH 24 MO	4000 FH 24 MO	ALL	ALL
	Lubricate the left wing flap skew sensor mechanism. **INTERVAL NOTE:** Whichever occurs first. **ACCESS NOTE:** Flaps deployed.						
27-136-02-01	MRB	12-22-51-640-807 12-22-51-640-808 12-22-51-640-809 12-22-51-640-810	1.1 1.2	4000 FH 24 MO	4000 FH 24 MO	ALL	ALL
	Lubricate the right wing flap skew sensor mechanism. **INTERVAL NOTE:** Whichever occurs first. **ACCESS NOTE:** Flaps deployed.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
27-138-00-01	MRB	12-22-51-640-801	1.1 1.2	4800 FC 24 MO	4800 FC 24 MO	ALL	ALL
	Lubricate the left wing flap drive torque tube supports and couplings. **INTERVAL NOTE:** Whichever occurs first. **ACCESS NOTE:** Flaps deployed.						
27-138-00-02	MRB	12-22-51-640-801	1.1 1.2	4800 FC 24 MO	4800 FC 24 MO	ALL	ALL
	Lubricate the right wing flap drive torque tube supports and couplings. **INTERVAL NOTE:** Whichever occurs first. **ACCESS NOTE:** Flaps deployed.						
27-140-01-01	MRB	12-22-51-610-803	1.1	5000 FC	5000 FC	ALL	ALL
	Check the left wing flap drive transmission oil level and service as required. **ACCESS NOTE:** Flaps deployed.						
27-140-02-01	MRB	12-22-51-610-803	1.1	5000 FC	5000 FC	ALL	ALL
	Check the right wing flap drive transmission oil level and service as required. **ACCESS NOTE:** Flaps deployed.						
27-142-01-01	MRB	12-22-51-610-804	1.1	25000 FC	25000 FC	ALL	ALL
	Replace the left wing flap drive transmission oil. **ACCESS NOTE:** Flaps deployed.						
27-142-02-01	MRB	12-22-51-610-804	1.1	25000 FC	25000 FC	ALL	ALL
	Replace the right wing flap drive transmission oil. **ACCESS NOTE:** Flaps deployed.						
27-144-00-01	MRB	12-22-51-640-802 12-22-51-640-803 12-22-51-640-804 12-22-51-640-805	1.1	1000 FC	1000 FC	ALL	ALL
	Lubricate the left wing trailing edge flap ballscrew assemblies and flap transmission universal joints. **ACCESS NOTE:** Flaps deployed.						
27-144-00-02	MRB	12-22-51-640-802 12-22-51-640-803 12-22-51-640-804 12-22-51-640-805	1.1	1000 FC	1000 FC	ALL	ALL
	Lubricate the right wing trailing edge flap ballscrew assemblies and flap transmission universal joints. **ACCESS NOTE:** Flaps deployed.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
27-148-01-01	MRB	27-51-32-220-801 27-51-32-220-802 27-51-42-220-801 27-51-42-220-802	1.1	25000 FC	25000 FC	ALL	ALL
	Functionally check the left wing trailing edge flaps ballscrew assembly backlash. **ACCESS NOTE:** Flaps deployed.						
27-148-02-01	MRB	27-51-32-220-801 27-51-32-220-802 27-51-42-220-801 27-51-42-220-802	1.1	25000 FC	25000 FC	ALL	ALL
	Functionally check the right wing trailing edge flaps ballscrew assembly backlash. **ACCESS NOTE:** Flaps deployed.						
27-148-03-01	MPD	27-51-32-200-803 27-51-32-200-804 27-51-42-200-801 27-51-42-200-802	1.1	13200 FC	6600 FC	ALL	ALL
	Detailed Inspection of the left wing trailing edge flap ballscrew actuator for grease leakage, wear and condition						
27-148-04-01	MPD	27-51-32-200-803 27-51-32-200-804 27-51-42-200-801 27-51-42-200-802	1.1	13200 FC	6600 FC	ALL	ALL
	Detailed Inspection of the right wing trailing edge flap ballscrew actuator for grease leakage, wear and condition						
27-152-01-01	MRB	12-22-51-640-806	1.1 1.2	2000 FC 12 MO	2000 FC 12 MO	ALL	ALL
	Lubricate the #4 flap transmission angle/tee gearbox universal joints. **INTERVAL NOTE:** Whichever comes first.						
27-152-02-01	MRB	12-22-51-640-806	1.1 1.2	2000 FC 12 MO	2000 FC 12 MO	ALL	ALL
	Lubricate the #5 flap transmission angle/tee gearbox universal joints. **INTERVAL NOTE:** Whichever comes first.						
27-154-00-01	MRB	27-51-00-740-803	1.1	5000 FH	5000 FH	ALL	ALL
	Operationally check the flap load relief system.						
27-156-00-01	MRB	27-51-00-720-804	1.1	25000 FH	25000 FH	ALL	ALL
	Functionally check the flap load relief system.						
27-158-00-01	MRB	27-51-00-720-803	1.1	15000 FH	15000 FH	ALL	ALL
	Functionally check the alternate flap drive system.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
27-162-00-01	MRB	27-51-00-740-802	1.1	750 FH	750 FH	ALL	ALL
	Operationally check the flap skew and flap asymmetry systems by initiating a BITE check of the Flap/Slat Electronics Unit (FSEU).						
27-164-00-01	MRB	27-51-00-740-801	1.1	15000 FH	15000 FH	ALL	ALL
	Operationally check the flap uncommanded motion protection system by initiating a bite check of the flap slat electronics unit.						
27-166-00-01	MRB	27-51-06-210-801	1.1	25000 FH	25000 FH	ALL	ALL
	Perform a detail visual inspection of the trailing edge flap lever sensor, linkage, rod and rod ends.						
27-168-01-01	MRB	27-51-00-710-802	1.1	25000 FC	25000 FC	ALL	ALL
	Operationally check the left wing trailing edge flap transmission no-back brakes. **ACCESS NOTE:** Flaps deployed.						
27-168-02-01	MRB	27-51-00-710-802	1.1	25000 FC	25000 FC	ALL	ALL
	Operationally check the right wing trailing edge flap transmission no-back brakes. **ACCESS NOTE:** Flaps deployed.						
27-170-01-01	MRB	12-22-51-640-811 12-22-51-640-812	1.1 1.2	1250 FC 8 MO	1250 FC 8 MO	ALL	ALL
	Lubricate the left wing trailing edge flap actuation mechanism. **INTERVAL NOTE:** Whichever occurs first. **ACCESS NOTE:** Flaps deployed.						
27-170-02-01	MRB	12-22-51-640-811 12-22-51-640-812	1.1 1.2	1250 FC 8 MO	1250 FC 8 MO	ALL	ALL
	Lubricate the right wing trailing edge flap actuation mechanism. **INTERVAL NOTE:** Whichever occurs first. **ACCESS NOTE:** Flaps extended.						
27-171-01-01	MRB	27-51-00-210-801	1.1	12000 FH	12000 FH	ALL	ALL
	Perform a general visual inspection of the left wing flap drive seal rib angle gear box, MLG beam angle gear box, flap drive torque tubes, torque tube couplings and support and flap transmissions. **ACCESS NOTE:** Flaps deployed.						
27-171-02-01	MRB	27-51-00-210-801	1.1	12000 FH	12000 FH	ALL	ALL
	Perform a general visual inspection of the right wing flap drive seal rib angle gear box, MLG beam angle gear box, flap drive torque tubes, torque tube couplings and support and flap transmissions. **ACCESS NOTE:** Flaps deployed.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
27-172-01-01	MRB	27-51-00-210-802	1.1	12000 FC	12000 FC	ALL	ALL
	Perform a detail visual inspection of the left wing trailing edge flap actuation mechanism. **ACCESS NOTE:** Flaps deployed.						
27-172-02-01	MRB	27-51-00-210-802	1.1	12000 FC	12000 FC	ALL	ALL
	Perform a detail visual inspection of the right wing trailing edge flap actuation mechanism. **ACCESS NOTE:** Flaps deployed.						
27-174-01-01	MRB	12-22-51-640-814 12-22-51-640-815 12-22-51-640-816	1.1	4000 FC	4000 FC	ALL	ALL
	Lubricate the left wing #'s 1, 2, and 3 trailing edge flap track forward attachment point pins.						
27-174-02-01	MRB	12-22-51-640-814 12-22-51-640-815 12-22-51-640-816	1.1	4000 FC	4000 FC	ALL	ALL
	Lubricate the right wing #'s 6, 7, and 8 trailing edge flap track forward attachment point pins.						
27-176-01-01	MRB	12-22-51-640-813	1.1	1000 FC	1000 FC	ALL	ALL
	Lubricate the left wing #4 inboard flap track attachment fittings.						
27-176-02-01	MRB	12-22-51-640-813	1.1	1000 FC	1000 FC	ALL	ALL
	Lubricate the right wing #5 inboard flap track attachment fittings.						
27-178-00-01	MRB	12-22-51-610-805	1.1	7500 FH	7500 FH	ALL	ALL
	Check alternate flap drive gearbox oil level and service as required.						
27-182-00-01	MRB	12-22-61-600-801	1.1	4000 FH	4000 FH	ALL	ALL
	Lubricate the spoiler mixer.						
27-182-01-01	MRB	12-22-61-600-802 12-22-61-640-801	1.1	4000 FH	4000 FH	ALL	ALL
	Lubricate the left wing spoiler mechanical control path. **ACCESS NOTE:** Flaps deployed.						
27-182-02-01	MRB	12-22-61-600-802 12-22-61-640-801	1.1	4000 FH	4000 FH	ALL	ALL
	Lubricate the right wing spoiler mechanical control path. **ACCESS NOTE:** Flaps deployed.						
27-184-00-01	MRB	27-61-00-820-809	1.1	22400 FH	22400 FH	ALL	ALL
	Functionally check the torque of the spoiler ratio changer no-back assembly.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
27-186-00-01	MRB	12-22-81-600-801	1.1	6000 FH	6000 FH	ALL	ALL
	Lubricate the speedbrake lever no-back brake.						
27-187-00-01	MSG3	27-62-00-710-802	1.1	6000 FH	6000 FH	ALL	ALL
	Operationally Check the Speedbrake Handle Stop **AIRPLANE NOTE:** Applicable to 900ER and airplanes with Short Field Performance Package (if installed).						
27-188-00-02	MRB	27-62-00-820-809	1.1	15000 FH	15000 FH	ALL	ALL
	Operationally check the speedbrake refused takeoff (RTO) system. Note: This task is applicable to airplanes with Short Field Performance Package (if installed).						
27-190-00-01	MRB	27-62-00-820-810	1.1	11000 FH	11000 FH	ALL	ALL
	Operationally check the speedbrakes extended light.						
27-192-01-01	MRB	27-61-00-820-810	1.1	25000 FH	25000 FH	ALL	ALL
	Perform an operational check of each left wing flight spoiler actuator override quandrant. **ACCESS NOTE:** Flaps deployed.						
27-192-02-01	MRB	27-61-00-820-810	1.1	25000 FH	25000 FH	ALL	ALL
	Perform an operational check of each right wing flight spoiler actuator override quadrant. **ACCESS NOTE:** Flaps deployed.						
27-194-00-01	MRB	27-61-00-710-802	1.1	20000 FH	20000 FH	ALL	ALL
	Operationally check the spoiler mixer centering mechanism. **SPECIAL NOTE:** CMR (27-CMR-04) interval for this task is 20,000 FH. See MPD Section 9.						
27-196-00-01	MSG3	27-62-00-760-801	1.1	7000 FH	7000 FH	ALL	ALL
	Functionally Check the Spoiler Electrical Control System Relays for Continuity **AIRPLANE NOTE:** Applicable to 900ER and airplanes with Short Field Performance Package (if installed).						
27-210-00-01	MRB	29-00-00-790-809	1.1	25000 FH	25000 FH	ALL	ALL
	Functionally check the internal leakage of leading edge slat actuators.						
27-212-00-01	MRB	29-00-00-790-809	1.1	25000 FH	25000 FH	ALL	ALL
	Functionally check the internal leakage of leading edge flap actuators.						
27-214-00-01	MRB	27-81-00-710-801	1.1	7500 FH	7500 FH	ALL	ALL
	Operationally check the leading edge standby actuation system.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
27-215-01-01	MRB	27-62-00-210-802	1.1	6000 FH	6000 FH	ALL	ALL
	Perform a general visual inspection of the left wing spoiler actuators.						
27-215-02-01	MRB	27-62-00-210-802	1.1	6000 FH	6000 FH	ALL	ALL
	Perform a general visual inspection of the right wing spoiler actuators.						
27-215-03-01	MRB	27-61-00-210-801	1.1	5000 FH	5000 FH	ALL	ALL
	Perform a general visual inspection of the spoiler mechanical control path.						
27-216-00-01	MRB	27-83-00-710-801	1.1	3000 FH	3000 FH	ALL	ALL
	Functionally check the autoslat system.						
27-218-00-01	MRB	27-81-00-700-804	1.1	5000 FH	5000 FH	ALL	ALL
	Operationally check the leading edge uncommanded motion protection system.						
27-220-01-01	MRB	12-22-71-600-801	1.1 1.2	1250 FC 8 MO	1250 FC 8 MO	ALL	ALL
	Lubricate the left wing leading edge slat rollers. **INTERVAL NOTE:** Whichever occurs first. **ACCESS NOTE:** Leading edges extended.						
27-220-02-01	MRB	12-22-71-600-801	1.1 1.2	1250 FC 8 MO	1250 FC 8 MO	ALL	ALL
	Lubricate the right wing leading edge slat rollers. **INTERVAL NOTE:** Whichever occurs first. **ACCESS NOTE:** Leading edges extended.						
27-222-01-01	MRB	12-22-71-640-801	1.1 1.2	2500 FC 16 MO	2500 FC 16 MO	ALL	ALL
	Lubricate the left wing leading edge slat tracks. **INTERVAL NOTE:** Whichever occurs first. **ACCESS NOTE:** Leading edges extended.						
27-222-02-01	MRB	12-22-71-640-801	1.1 1.2	2500 FC 16 MO	2500 FC 16 MO	ALL	ALL
	Lubricate the right wing leading edge slat tracks. **INTERVAL NOTE:** Whichever occurs first. **ACCESS NOTE:** Leading edges extended.						
27-224-00-01	MRB	27-81-00-710-802	1.1	1250 FH	1250 FH	ALL	ALL
	Operationally check the leading edge devices uncommanded motion protection using the standby hydraulic system.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
27-225-01-01	MRB	27-81-00-210-801	1.1	6000 FH	6000 FH	ALL	ALL

Perform a general visual inspection of the left wing leading edge flap and slat actuators and left wing leading edge flap and slat actuation mechanisms.

ACCESS NOTE: Leading edges extended.

27-225-02-01	MRB	27-81-00-210-801	1.1	6000 FH	6000 FH	ALL	ALL

Perform a general visual inspection of the right wing leading edge flap and slat actuators and right wing leading edge flap and slat actuation mechanisms.

ACCESS NOTE: Leading edges extended.

27-226-00-01	MRB	20-20-31-200-801 20-20-31-200-802 20-20-31-200-805	1.1 1.2	4000 FC 24 MO	4000 FC 24 MO	ALL	ALL

Perform a detail visual inspection of the control cables within the left main landing gear wheel well for broken wires. Check associated pulleys, brackets, and mechanisms for condition and security of installation. The following cables are located in the left MLG wheel well:

A. Aileron control cables

B. Spoiler control cables

C. Speed brake control cables

Note: The control cables must be displaced full travel in each direction for complete inspection at seals, pulleys, and fairlead areas.

INTERVAL NOTE: Whichever occurs first.

27-226-00-02	MRB	20-20-31-200-801 20-20-31-200-802 20-20-31-200-805	1.1 1.2	4000 FC 24 MO	4000 FC 24 MO	ALL	ALL

Perform a detail visual inspection of the control cables within the right main landing gear wheel well for broken wires. Check associated pulleys, brackets, and mechanisms for condition and security of installation. The following cables are located within the right MLG wheel well:

A. Aileron Cables

B. Flap control cables

C. Spoiler control cables

D. Speed brake control cables

Note: The control cables must be displaced full travel in each direction for complete inspection at seals, pulleys, and fairlead areas.

INTERVAL NOTE: Whichever occurs first.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
27-226-00-03	MRB	20-20-31-200-801 20-20-31-200-802 20-20-31-200-805	1.1 1.2	4000 FC 24 MO	4000 FC 24 MO	ALL	ALL

Perform a detail visual inspection of the control cables within the left wing aft spar area for broken wires. Check associated pulleys, brackets, and mechanisms for condition and security of installation. The following cables are located within the left wing aft spar area:

A. Aileron control cables

B. Wing spoiler control cables

Note: The control cables must be displaced full travel in each direction for complete inspection at seals, pulleys, and fairlead areas.

INTERVAL NOTE: Whichever occurs first.

ACCESS NOTE: Extend Flaps

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
27-226-00-04	MRB	20-20-31-200-801 20-20-31-200-802 20-20-31-200-805	1.1 1.2	4000 FC 24 MO	4000 FC 24 MO	ALL	ALL

Perform a detail visual inspection of the control cables within the right wing aft spar area for broken wires. Check associated pulleys, brackets, and mechanisms for condition and security of installation. The following cables are located within the right wing aft spar area:

A. Aileron control cables

B. Wing spoiler control cables

Note: The control cables must be displaced full travel in each direction for complete inspection at seals, pulleys, and fairlead areas.

INTERVAL NOTE: Whichever occurs first.

ACCESS NOTE: Extend Flaps

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
27-228-00-01	MRB	20-20-31-200-801 20-20-31-200-802 20-20-31-200-805	1.1 1.2	6600 FC 3 YR	6600 FC 3 YR	ALL	ALL

Perform a detail visual inspection of all internal portions of the flight control cables above the MLG wheel well from B.S. 663.75 to B.S. 727 for broken wires, associated pulleys, brackets, and mechanisms for condition and security of installation.

Note: The control cable system must be displaced full travel in each direction for complete inspection at seals, pulleys, and fairlead areas.

INTERVAL NOTE: Whichever occurs first.

ACCESS NOTE: Passenger cabin floor panels between B.S. 663.75 and B.S. 727

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
27-228-00-02	MRB	20-20-31-200-801 20-20-31-200-802 20-20-31-200-805	1.1 1.2	6600 FC 3 YR	6600 FC 3 YR	ALL	ALL

Perform a detail visual inspection of all internal portions of the flight control cables for broken wires, associated pulleys, brackets, and mechanisms for condition and security of installation.

Note: The control cable system must be displaced full travel in each direction for complete inspection at seals, pulleys, and fairlead areas.

INTERVAL NOTE: Whichever occurs first.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
27-228-00-03	MRB	20-20-31-200-801 20-20-31-200-802 20-20-31-200-805	1.1 1.2	6600 FC 3 YR	6600 FC 3 YR	ALL	ALL

Perform a detail visual inspection of all internal portions of the flight control cables for broken wires within the electronics compartment for associated pulleys, brackets, and mechanisms for condition and security of installation.
Note: The control cable system must be displaced full travel in each direction for complete inspection at seals, pulleys, and fairlead areas.

INTERVAL NOTE: Whichever occurs first.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
27-228-00-04	MRB	20-20-31-200-801 20-20-31-200-802 20-20-31-200-805	1.1 1.2	6600 FC 3 YR	6600 FC 3 YR	ALL	ALL

Perform a detail visual inspection of all internal portions of the flight control cables for broken wires within the air conditioning distribution bay, associated pulleys, brackets, and mechanisms for condition and security of installation.
Note: The control cable system must be displaced full travel in each direction for complete inspection at seals, pulleys, and fairlead areas.

INTERVAL NOTE: Whichever occurs first.

ACCESS NOTE: Forward Cargo Compartment Aft Bulkhead.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
27-228-00-05	MRB	20-20-31-200-801 20-20-31-200-802 20-20-31-200-805	1.1 1.2	6600 FC 3 YR	6600 FC 3 YR	ALL	ALL

Perform a detail visual inspection of all internal portions of the flight control cables for broken wires within the forward cargo compartment, associated pulleys, brackets, and mechanisms for condition and security of installation.
Note: The control cable system must be displaced full travel in each direction for complete inspection at seals, pulleys, and fairlead areas.

INTERVAL NOTE: Whichever occurs first.

ACCESS NOTE: Forward Cargo Compartment Ceiling Panels or Floor Panels between B.S.396 to B.S. 540.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
27-228-00-06	MRB	20-20-31-200-801 20-20-31-200-802 20-20-31-200-805	1.1 1.2	6600 FC 3 YR	6600 FC 3 YR	ALL	ALL

Perform a detail visual inspection of all internal portions of the flight control cables for broken wires within the aft cargo compartment, associated pulleys, brackets, and mechanisms for condition and security of installation.
Note: The control cable system must be displaced full travel in each direction for complete inspection at seals, pulleys, and fairlead areas.

INTERVAL NOTE: Whichever occurs first.

ACCESS NOTE: Aft Cargo Compartment Ceiling Panels

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
27-228-00-07	MRB	20-20-31-200-801 20-20-31-200-802 20-20-31-200-805	1.1 1.2	6600 FC 3 YR	6600 FC 3 YR	ALL	ALL

Perform a detail visual inspection of all internal portions of the flight control cables for broken wires within the aft cargo equipment bay, associated pulleys, brackets, and mechanisms for condition and security of installation.
Note: The control cable system must be displaced full travel in each direction for complete inspection at seals, pulleys, and fairlead areas.

INTERVAL NOTE: Whichever occurs first.

ACCESS NOTE: Aft cargo compartment aft bulkhead panels and water tank, or the pressurization aft outflow valve assembly.

27-228-00-08	MRB	20-20-31-200-801 20-20-31-200-802 20-20-31-200-805	1.1 1.2	6600 FC 3 YR	6600 FC 3 YR	ALL	ALL

Perform a detail visual inspection of all internal portions of the flight control cables for broken wires within the tail compartment, associated pulleys, brackets, and mechanisms for condition and security of installation.
Note: The control cable system must be displaced full travel in each direction for complete inspection at seals, pulleys, and fairlead areas.

INTERVAL NOTE: Whichever occurs first.

27-229-00-01	MRB	20-20-31-200-801 20-20-31-200-802	1.1 1.2	21600 FC 6 YR	21600 FC 6 YR	ALL	ALL

Inspect (detailed) inboard trailing edge aft flap drive cable.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Flaps extended.

27-229-00-02	MRB	20-20-31-200-801 20-20-31-200-802	1.1 1.2	21600 FC 6 YR	21600 FC 6 YR	ALL	ALL

Inspect (detailed) inboard trailing edge aft flap drive cable.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Flaps extended.

27-230-00-01	MRB	20-20-31-200-801 20-20-31-200-802 20-20-31-200-805	1.1 1.2	36000 FC 12 YR	36000 FC 12 YR	ALL	ALL

Perform a detail visual inspection of all flight control cables for broken wires within the passenger compartment over the wing center section from B.S. 540 to B.S. 663.75. Check associated pulleys, brackets, and mechanisms for condition and security of installation.
Note: The control cable system must be displaced full travel in each direction for complete inspection at seals, pulleys, and fairlead areas.

INTERVAL NOTE: Whichever occurs first.

ACCESS NOTE: Passenger cabin floor panels between B.S. 540 to B.S. 663.75.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
27-230-00-02	MRB	20-20-31-200-801 20-20-31-200-802 20-20-31-200-805	1.1 1.2	36000 FC 12 YR	36000 FC 12 YR	ALL	ALL
	Perform a detail visual inspection of all flight control cables for broken wires within the pilot's control quadrant. Check associated pulleys, brackets, and mechanisms for condition and security of installation. Note: The control cable system must be displaced full travel in each direction for complete inspection at seals, pulleys, and fairlead areas. **INTERVAL NOTE:** Whichever occurs first.						
27-235-00-02	MRB	27-11-00-820-801 27-21-00-820-808-002 27-31-00-820-801	1.1 1.2	6600 FC 3 YR	6600 FC 3 YR	ALL	ALL
	Functionally check flight control cable tension. (Airplanes with the rudder system enhancement) **INTERVAL NOTE:** Whichever comes first.						
28-010-00-01	MRB	28-11-00-210-801	1.1	8000 FH	8000 FH	ALL	ALL
	Inspect (general visual) the main and center fuel tank (wing) external lower surfaces including tank vents, sump drain valves, skin lap joints for obvious leaks, condition, and security.						
28-020-01-01	MRB	28-13-41-200-801	1.1	3 YR	3 YR	ALL	ALL
	Operationally check the left surge tank pressure relief valves.						
28-020-02-01	MRB	28-13-41-200-801	1.1	3 YR	3 YR	ALL	ALL
	Operationally check the right surge tank pressure relief valves.						
28-030-01-01	MRB	28-13-31-100-801	1.1	3 YR	3 YR	ALL	ALL
	Inspect (detailed) the left surge tank vent flame arrestor for clogging, condition, and security.						
28-030-02-01	MRB	28-13-31-100-801	1.1	3 YR	3 YR	ALL	ALL
	Inspect (detailed) the right surge tank vent flame arrestor for clogging, condition, and security.						
28-040-00-01	MRB	28-22-00-720-802	1.1	24000 FH	24000 FH	ALL	ALL
	Operationally check the center tank fuel scavenge system (jet pump, induced port check valve, float valve, and line).						
28-050-00-01	MRB	28-22-00-710-802	1.1	7500 FH	7500 FH	ALL	ALL
	Operationally check engine fuel suction feed system. **SPECIAL NOTE:** AWL task (28-AWL-101) interval for this task is 7500 FH or 3 YRS, whichever occurs first. See MPD section 9.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
28-054-00-01	MRB	28-22-41-720-802	1.1	7500 FH	7500 FH	ALL	ALL
	Operationally (BITE) check the fuel pump GFI relay **SPECIAL NOTE:** AWL task (28-AWL-20) interval for this task is 1 YR. See MPD section 9. **AIRPLANE NOTE:** Applicable to airplanes line number 1981 and 2093. Applicable to airplanes line number 2210 and on. Applicable to airplanes line number 1 to 1980, 1982 to 2092 and 2094 to 2209 which have incorporated SB 737 28A1201.						
28-056-00-01	MRB	28-22-00-720-806	1.1	12000 FH	12000 FH	ALL	ALL
	Functionally check Center Tank Fuel Boost Pump Power Failed On Protection System. **SPECIAL NOTE:** AWL task (28-AWL-23) interval for this task is 1 YR. See MPD section 9. **AIRPLANE NOTE:** Applicable to airplane line number 1973 and on. Applicable to airplane line number 1 to 1972 incorporating SB 737 28A1248.						
28-060-01-01	MRB	28-22-13-200-802	1.1	4000 FH	4000 FH	ALL	ALL
	Restore (clean) the left main fuel tank water scavenge jet pump (without defueling the tank).						
28-060-02-01	MRB	28-22-13-200-802	1.1	4000 FH	4000 FH	ALL	ALL
	Restore (clean) the right main fuel tank water scavenge jet pump (without defueling the tank).						
28-060-03-01	MRB	28-22-13-200-801	1.1	4000 FH	4000 FH	ALL	ALL
	Restore (clean) the left and right water scavenge jet pumps in the center wing fuel tank (without defueling the tank).						
28-070-00-01	MRB	28-25-05-790-801	1.1	12000 FH	12000 FH	ALL	ALL
	Functionally (pressure decay) check the APU fuel line shroud.						
28-080-00-01	MRB	28-41-00-710-801	1.1	15000 FH	15000 FH	ALL	ALL
	Operationally (BITE) check the fuel quantity indicating system.						
28-090-00-01	ALI	28-22-00-200-802	1.1	10 YR	10 YR	ALL	ALL
	Perform a detailed inspection of the out of tank wire bundles installed on specified brackets that are mounted directly on the fuel tanks. **SPECIAL NOTE:** AWL task (28-AWL-29) interval for this task is 10 YR. See MPD section 9.						
28-115-00-01	ALI	28-22-00-720-805	1.1	1 YR	1 YR	ALL	ALL
	Functionally check the center tank boost pump auto shutoff system. **SPECIAL NOTE:** AWL task (28-AWL-19) interval for this task is 1 YR. See MPD section 9. **AIRPLANE NOTE:** Applicable to airplane line numbers 1494 and on, and operators that have incorporated Boeing Service Bulletin 737-28A1206. **INTERVAL NOTE:** 1 YR interval is from 28-AWL-19. MSG-3 interval is 4 years or 15000 hours, whichever comes first.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
28-125-00-01	MRB	28-22-00-720-801	1.1	15000 FH	15000 FH	ALL	ALL
	Operationally check the fuel shutoff valve battery.						
28-130-00-01	MRB	28-22-14-000-801 28-22-14-400-801	1.1	7 YR	7 YR	ALL	ALL
	Replace the fuel shut-off valve battery.						
28-140-00-01	MRB	28-00-00-280-801	1.1	10 YR	10 YR	ALL	ALL
	Inspect (detailed) in-tank tubing and equipment static ground straps and clamps for condition, security and other degradation. (SFAR 88) **INTERVAL NOTE:** Perform concurrently with other tank inspections to minimize tank entries and possible accidental damage.						
28-140-00-02	MRB	28-00-00-280-802	1.1	10 YR	10 YR	ALL	ALL
	Inspect (detailed) in-tank tubing and equipment static ground straps and clamps for condition, security and other degradation. (SFAR 88) **INTERVAL NOTE:** Perform concurrently with other tank inspections to minimize tank entries and possible accidental damage.						
28-140-00-03	MRB	28-00-00-280-803	1.1	10 YR	10 YR	ALL	ALL
	Inspect (detailed) in-tank tubing and equipment static ground straps and clamps for condition, security and other degradation. (SFAR 88) **INTERVAL NOTE:** Perform concurrently with other tank inspections to minimize tank entries and possible accidental damage.						
28-150-00-01	MRB	28-00-00-760-801	1.1	6 YR	6 YR	ALL	ALL
	Perform a functional check (resistance measurement) the bonding between fuel pumps and adjoining structure for the left and center pumps located external to the left wing fuel tank. (SFAR 88)						
28-150-00-02	MRB	28-00-00-760-802	1.1	6 YR	6 YR	ALL	ALL
	Perform a functional check (resistance measurement) the bonding between fuel pumps and adjoining structure for the right and center pumps located external to the right wing fuel tank. (SFAR 88)						
28-160-00-01	MRB	28-00-00-760-803	1.1	6 YR	6 YR	ALL	ALL
	Functionally check (resistance measurement) the bonding resistance between the APU DC Fuel Pump motor bonding strap (if APU DC Fuel Pump installed).(SFAR 88) **AIRPLANE NOTE:** If the APU fuel boost pump is installed.						
28-170-00-01	MRB	20-60-03-100-801 28-41-44-280-801	1.1	10 YR	10 YR	ALL	ALL
	Inspect (detailed) the in-tank FQIS exposed Electrical Wiring Interconnection System (EWIS) and EWIS support for damage, adequate separation with structure and proper security. (SFAR 88) (EZAP) **INTERVAL NOTE:** Perform task concurrently with other fuel tank inspection tasks to minimize tank entries and possible accidental damage.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
28-170-00-02	MRB	20-60-03-100-801 28-41-44-280-802	1.1	10 YR	10 YR	ALL	ALL
	Inspect (detailed) the in-tank FQIS exposed Electrical Wiring Interconnection System (EWIS) and EWIS support for damage, adequate separation with structure and proper security. (SFAR 88) (EZAP) **INTERVAL NOTE:** Perform task concurrently with other fuel tank inspection tasks to minimize tank entries and possible accidental damage.						
28-170-00-03	MRB	20-60-03-100-801 28-41-44-280-803	1.1	10 YR	10 YR	ALL	ALL
	Inspect (detailed) the in-tank FQIS exposed Electrical Wiring Interconnection System (EWIS) and EWIS support for damage, adequate separation with structure and proper security. (SFAR 88) (EZAP) **INTERVAL NOTE:** Perform task concurrently with other fuel tank inspection tasks to minimize tank entries and possible accidental damage.						
28-171-00-01	MRB	28-41-44-280-801	1.1	10 YR	10 YR	ALL	ALL
	Inspect (detailed) the in-tank FQIS components for chaffing, rubbing, adequate separation from structure and condition for security. (SFAR 88)						
28-171-00-02	MRB	28-41-44-280-802	1.1	10 YR	10 YR	ALL	ALL
	Inspect (detailed) the in-tank FQIS components for chaffing, rubbing, adequate separation from structure and condition for security. (SFAR 88)						
28-171-00-03	MRB	28-41-44-280-803	1.1	10 YR	10 YR	ALL	ALL
	Inspect (detailed) the in-tank FQIS components for chaffing, rubbing, adequate separation from structure and condition for security. (SFAR 88)						
28-173-00-01	MRB	05-55-54-200-801	1.1	12 YR	12 YR	ALL	ALL
	Functionally check (resistance measurement) out tank FQIS wire bundle lightning shield to ground termination. (SFAR 88) **SPECIAL NOTE:** AWL task (28-AWL-03) interval for this task is 10 YRS. See MPD section 9.						
28-200-00-01	ALI	28-11-00-211-801	1.1	10 YR	10 YR	ALL	ALL
	Perform a detailed inspection of the wire bundles routed on the main deck over the center fuel tank and under the main deck floor boards. **SPECIAL NOTE:** AWL task (28-AWL-01) interval for this task is 10 YR. See MPD section 9.						
28-201-00-01	MRB	29-11-04-200-801	1.1	12 YR	12 YR	ALL	ALL
	Functionally check (resistance measurement) the bonding resistance between the hydraulic line fitting at fuel tank wall penetrations and the adjoining structure. (SFAR 88)						
28-204-00-01	MRB	28-13-31-200-801	1.1	12 YR	12 YR	ALL	ALL
	Functionally check (resistance measurement) the bonding between the lower air vent stack and the door structure. (SFAR 88)						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
28-205-00-01	MRB	28-13-41-200-802	1.1	12 YR	12 YR	ALL	ALL
	Functionally check (resistance measurement) the bonding between the pressure relief valve and the structure. (SFAR 88)						
28-207-00-01	MRB	28-21-51-200-801	1.1	12 YR	12 YR	ALL	ALL
	Functionally check (resistance measurement) the bonding of the fueling shutoff valve actuator (solenoid) to the adjoining structure. (SFAR 88)						
28-208-00-01	MRB	28-21-11-200-801	1.1	12 YR	12 YR	ALL	ALL
	Functionally check (resistance measurement) the bonding between the fueling receptacle (manifold) and the structure. (SFAR 88)						
28-211-00-01	MRB	28-22-11-200-801	1.1	12 YR	12 YR	ALL	ALL
	Functionally check (resistance measurement) the bonding between the engine spar motor operated valve actuator and adjoining structure. (SFAR 88) Note: This task is performed from outside of tank.						
28-211-00-02	MRB	28-22-21-200-801	1.1	12 YR	12 YR	ALL	ALL
	Functionally check (resistance measurement) the bonding between the fuel crossfeed motor operated valve actuator and adjoining structure. (SFAR 88) Note: This task is performed from outside of tank.						
28-213-00-01	MRB	28-25-02-200-801	1.1	12 YR	12 YR	ALL	ALL
	Functionally check (resistance measurement) the bonding resistance between the APU shutoff valve actuator and adjoining structure. (SFAR 88)						
29-010-00-01	MRB	29-00-00-790-802	1.1	24000 FH	24000 FH	ALL	ALL
	Gross internal hydraulic system leakage check.						
29-020-00-01	MRB	29-11-71-000-802 29-11-71-400-802	1.1	8000 FH	8000 FH	ALL	ALL
	Replace the hydraulic system "A" pressure filter elements for electric motor driven pumps (EMDP).						
29-020-00-02	MRB	29-11-71-000-802 29-11-71-400-802	1.1	8000 FH	8000 FH	ALL	ALL
	Replace the hydraulic system "B" pressure filter elements for electric motor driven pumps (EMDP).						
29-030-01-01	MRB	29-11-41-000-801 29-11-41-400-801	1.1	600 FH	600 FH	ALL	ALL
	Replace the "A" hydraulic system electric motor driven pump (EMDP) case drain filter.						
29-030-02-01	MRB	29-11-41-000-801 29-11-41-400-801	1.1	600 FH	600 FH	ALL	ALL
	Replace the "B" hydraulic system electric motor driven pump (EMDP) case drain filter.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
29-040-00-01	MRB	29-11-71-000-802 29-11-71-400-802	1.1	8000 FH	8000 FH	ALL	ALL
	Replace the hydraulic system "A" pressure filter elements for engine driven pumps (EDP).						
29-040-00-02	MRB	29-11-71-000-802 29-11-71-400-802	1.1	8000 FH	8000 FH	ALL	ALL
	Replace the hydraulic system "B" pressure filter elements for engine driven pumps (EDP).						
29-050-01-01	MRB	29-11-51-000-801 29-11-51-400-801	1.1	2400 FH	2400 FH	ALL	ALL
	Replace the "A" system EDP case drain filters.						
29-050-02-01	MRB	29-11-51-000-801 29-11-51-400-801	1.1	2400 FH	2400 FH	ALL	ALL
	Replace the "B" system EDP case drain filters.						
29-070-00-02	MRB	29-09-01-000-803 29-09-01-400-803	1.1	4000 FH	4000 FH	ALL	ALL
	Clean the reservoir pressurization filter assembly. Airplane note: This task is applicable to airplane line number 1345 and on, or line number 1-1344 that have incorporated Boeing Service Bulletin 737-29-1106.						
29-080-00-01	MRB	29-11-21-700-802	1.1	5000 FH	5000 FH	ALL	ALL
	Operational check of EMDP ground fault protection system.						
29-090-00-01	MRB	29-11-61-210-801	1.1	600 FH	600 FH	ALL	ALL
	Inspect (General Visual) the Delta "P" indication of A & B system return filter module.						
29-100-00-01	MRB	29-11-81-710-801	1.1	7500 FH	7500 FH	ALL	ALL
	Operational check of the A & B system engine pump (EDP) shutoff valves.						
29-110-00-01	MRB	32-33-71-000-803 32-33-71-400-802	1.1	25000 FH	25000 FH	ALL	ALL
	Remove the nose landing gear (NLG) down line fuse for functional test off aircraft.						
29-120-00-01	MRB	32-33-71-000-801 32-33-71-400-801	1.1	25000 FH	25000 FH	ALL	ALL
	Remove the nose landing gear (NLG) up line fuse for functional check off aircraft.						
29-130-00-01	MRB	F78-34-07-000-803-F00 F78-34-07-400-803-F00	1.1	25000 FH	25000 FH	ALL	ALL
	Remove the thrust reverser "A" system EMDP fuse for functional check off aircraft.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
29-140-00-01	MRB	27-81-61-000-801 27-81-61-400-801	1.1	25000 FH	25000 FH	ALL	ALL
	Remove the leading edge B system fuses for functional check off aircraft.						
29-150-00-01	MRB	32-41-72-000-801 32-41-72-400-801	1.1	25000 FH	25000 FH	ALL	ALL
	Remove the B system brake fuses for functional check off aircraft.						
29-160-00-01	MRB	32-41-72-020-801 32-41-72-420-801	1.1	25000 FH	25000 FH	ALL	ALL
	Remove the A system brake fuses for functional check off aircraft.						
29-170-00-01	MRB	27-21-95-000-801 27-21-95-400-801	1.1	25000 FH	25000 FH	ALL	ALL
	Remove the B system rudder fuse for functional check off aircraft.						
29-180-00-01	MRB	29-18-11-000-802 29-18-11-400-802	1.1	16000 FH	16000 FH	ALL	ALL
	Replace the ground hydraulic reservoir fill filter.						
29-200-00-01	MRB	29-21-00-700-801	1.1	15000 FH	15000 FH	ALL	ALL
	Operational check standby hydraulic electric motor driven pump to include observing low pressure light illumination and then going out.						
29-210-00-01	MRB	29-21-51-000-802 29-21-51-400-802	1.1	12000 FH	12000 FH	ALL	ALL
	Replace the standby hydraulic system pressure filter.						
29-220-00-01	MRB	29-21-41-000-801 29-21-41-400-801	1.1	12000 FH	12000 FH	ALL	ALL
	Replace the standby hydraulic system case drain filter.						
29-230-00-01	MRB	29-21-00-700-803	1.1 1.2	1200 FH 180 DY	1200 FH 180 DY	ALL	ALL
	Operationally check the standby rudder system. To include observing low pressure light illumination and then extinguishing. **INTERVAL NOTE:** Whichever occurs first.						
29-240-00-01	MRB	27-81-00-860-802	1.1	7500 FH	7500 FH	ALL	ALL
	Operationally check the alternate leading edge flaps and slats system. Note: This task is satisfied by accomplishment of task 27-214-00.						
29-250-00-01	MRB	29-22-00-710-801	1.1	7500 FH	7500 FH	ALL	ALL
	Operationally check the power transfer unit control system and check the control valve for proper position.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
29-260-00-01	MRB	29-22-00-730-801	1.1	25000 FH	25000 FH	ALL	ALL
	Functionally check the power transfer unit.						
29-270-00-01	MRB	29-22-21-020-801 29-22-21-400-802	1.1	12000 FH	12000 FH	ALL	ALL
	Replace the power transfer unit pressure filter.						
29-290-00-01	MRB	27-81-61-000-801 27-81-61-400-801	1.1	25000 FH	25000 FH	ALL	ALL
	Remove the leading edge standby hydraulic fuse for functional check off aircraft.						
29-300-00-01	MRB	F78-34-07-000-802-F00 F78-34-07-400-802-F00	1.1	25000 FH	25000 FH	ALL	ALL
	Remove the standby thrust reverser hydraulic fuse for functional check off aircraft.						
29-310-00-01	MRB	29-32-00-730-802 29-32-00-730-803	1.1	25000 FH	25000 FH	ALL	ALL
	Functionally check the A & B hydraulic system case drain warning switches. Note: Two AMM procedures are provided in this task card: AMM Task 29-32-00-730-802 (Preferred Method) and Task 29-32-00-730-803 (Alternate Method). Operators need to perform only one of the two procedures by selecting the one that would align with their maintenance practices.						
30-010-00-01	MRB	30-31-00-750-801	1.1	11000 FH	11000 FH	ALL	ALL
	Operationally check automatic activation of the Air Data Sensor heating for system A and B, if installed. NOTE: The AMM task provides a standard and an alternate procedure. The alternate method does not require engines running, a breakout box is used in place of the Display Electronics Unit 1. **AIRPLANE NOTE:** Applicable to airplanes with automatic Air Data Sensor heating installed. Airplanes Line Number 3424 and on, and airplanes incorporating SB 737-30A1063.						
31-010-00-01	MRB	31-51-00-740-801	1.1	15000 FH	15000 FH	ALL	ALL
	Functional check of the aural warning module (AWM) using bite check.						
31-020-00-01	MRB	21-33-00-000-801	1.1	6000 FH	6000 FH	ALL	ALL
	Functional check of the cabin pressure switch.						
31-030-00-01	MRB	34-16-00-730-801	1.1	15000 FH	15000 FH	ALL	ALL
	Operational check of mach/airspeed system 1 and 2 discrete outputs for aural warning system.						
31-040-00-01	MRB	31-51-00-730-802	1.1	6000 FH	6000 FH	ALL	ALL
	Operational check of landing gear logic module output to the AWM System 1 and 2.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
31-050-00-01	MRB	31-51-00-730-803	1.1	6000 FH	6000 FH	ALL	ALL
	Functional check of the take off warning system. This task tests: Trailing edge and leading edge flap, thrust lever, upper and lower stabilizer trim limit, ground spoiler pressure, speed brake, park brake and ground spoiler bypass valve switches for the aural warning module (AWM).						
31-120-00-04	MRB	31-31-00-700-801 31-31-00-970-807	1.1	7500 FH	7500 FH	ALL	ALL
	Download data from flight data recorder (FDR) to check interfacing system output to FDR (off aircraft).						
31-120-00-05	MRB	31-31-00-700-801 31-31-00-970-808	1.1	7500 FH	7500 FH	ALL	ALL
	Download data from flight data recorder (FDR) to check interfacing system output to FDR (off aircraft).						
31-130-00-04	MRB	31-31-00-970-807	1.1	7500 FH	7500 FH	ALL	ALL
	Functional check of required parameters (FDR, DFDAU output)						
31-130-00-05	MRB	31-31-00-970-808	1.1	7500 FH	7500 FH	ALL	ALL
	Functional check of required parameters (FDR, DFDAU output)						
31-140-00-01	MRB	31-31-09-000-801 31-31-09-400-801 31-31-09-700-801 31-31-09-700-802 31-31-09-960-803 31-31-09-960-804	1.1	NOTE		ALL	ALL
	Operational check of the ULB at battery replacement. **INTERVAL NOTE:** At battery replacement or national requirement. **ACCESS NOTE:** FDR Hinged Ceiling Panel.						
31-150-00-01	MRB	31-31-09-000-801 31-31-09-400-801 31-31-09-960-801 31-31-09-960-802	1.1	NOTE		ALL	ALL
	Replace ULB battery at vendor's recommendation. **INTERVAL NOTE:** At vendor's recommendation or national requirement. **ACCESS NOTE:** FDR Hinged Ceiling Panel.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
31-160-01-01	MRB	31-62-11-000-801 31-62-11-100-803 31-62-11-400-801	1.1	6000 FH	6000 FH	ALL	ALL
	Restore (Clean) all lint, dust and debris from the ventilation holes located on the aft side of the captain's primary flight, navigation, and center lower engine display units. **INTERVAL NOTE:** Display unit lint, dust and debris accumulation rates are dependent on operator environment, route structure and time of the year. Airline operators are encouraged to evaluate their particular 737NG operating environment, accompanied with debris findings at time of cleaning to identify and implement the most effective and economic maintenance interval. Operators should negotiate with their local regulatory agency to adjust intervals to a best fit for their individual operational environments.						
31-160-02-01	MRB	31-62-11-000-801 31-62-11-100-803 31-62-11-400-801	1.1	3000 FH	3000 FH	ALL	ALL
	Restore (Clean) all lint, dust and debris from the ventilation holes located on the aft side of the first officer's primary flight, navigation, and upper center engine display units. **INTERVAL NOTE:** Display unit lint, dust and debris accumulation rates are dependent on operator environment, route structure and time of the year. Airline operators are encouraged to evaluate their particular 737NG operating environment, accompanied with debris findings at time of cleaning to identify and implement the most effective and economic maintenance interval. Operators should negotiate with their local regulatory agency to adjust intervals to a best fit for their individual operational environments.						
32-010-01-01	MRB	32-00-10-100-802	1.1	50 FC	50 FC	ALL	ALL
	Clean exposed surfaces of the left main landing gear shock strut.						
32-010-02-01	MRB	32-00-10-100-802	1.1	50 FC	50 FC	ALL	ALL
	Clean exposed surfaces of the right main landing gear shock strut.						
32-020-01-01	MRB	12-15-31-610-802 12-15-31-610-805	1.1	4000 FC	4000 FC	ALL	ALL
	Service the left main landing gear shock strut. Servicing can be performed either with airplane on the ground or on jacks.						
32-020-02-01	MRB	12-15-31-610-802 12-15-31-610-805	1.1	4000 FC	4000 FC	ALL	ALL
	Service the right main landing gear shock strut. Servicing can be performed either with airplane on the ground or on jacks.						
32-030-01-01	MRB	12-21-11-640-801 12-21-11-640-802 12-25-07-600-801	1.1 1.2	560 FC 90 DY	560 FC 90 DY	ALL	ALL
	Lubricate the left main landing gear assembly. **INTERVAL NOTE:** Whichever comes first.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
32-030-02-01	MRB	12-21-11-640-801 12-21-11-640-802 12-25-07-600-801	1.1 1.2	560 FC 90 DY	560 FC 90 DY	ALL	ALL
	Lubricate the right main landing gear assembly. **INTERVAL NOTE:** Whichever comes first.						
32-040-01-01	MRB	32-11-00-000-801 32-11-00-400-801 32-11-61-000-803 32-11-61-400-803 32-11-71-000-801 32-11-71-420-801 32-11-83-000-801 32-11-83-400-801 32-11-89-000-801 32-11-89-420-801 32-32-11-000-801 32-32-11-400-801 32-32-31-000-801 32-32-31-400-801 57-16-01-000-801 57-16-01-400-801 57-16-02-000-801 57-16-02-400-801	1.1 1.2	21000 FC 10 YR	21000 FC 10 YR	ALL	ALL
	Restore the left main landing gear assembly. **INTERVAL NOTE:** Whichever comes first.						
32-040-02-01	MRB	32-11-00-000-801 32-11-00-400-801 32-11-61-000-803 32-11-61-400-803 32-11-71-000-801 32-11-71-420-801 32-11-83-000-801 32-11-83-400-801 32-11-89-000-801 32-11-89-420-801 32-32-11-000-801 32-32-11-400-801 32-32-31-000-801 32-32-31-400-801 57-16-01-000-801 57-16-01-400-801 57-16-02-000-801 57-16-02-400-801	1.1 1.2	21000 FC 10 YR	21000 FC 10 YR	ALL	ALL
	Restore the right main landing gear assembly. **INTERVAL NOTE:** Whichever comes first.						
32-050-01-01	MRB	32-11-00-000-801 32-11-00-400-801	1.1	75000 FC	75000 FC	ALL	ALL
	Discard the left main landing gear life limited parts.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
32-050-02-01	MRB	32-11-00-000-801 32-11-00-400-801	1.1	75000 FC	75000 FC	ALL	ALL
	Discard the right main landing gear life limited parts.						
32-060-00-01	MRB	32-00-10-100-802	1.1	50 FC	50 FC	ALL	ALL
	Clean exposed surface of the nose landing gear strut.						
32-070-00-01	MRB	12-15-41-610-802 12-15-41-610-805	1.1	3000 FC	3000 FC	ALL	ALL
	Service the nose landing gear shock strut. Servicing can be performed either with airplane on the ground or on jacks.						
32-080-00-01	MRB	12-21-21-640-801 12-21-21-640-802	1.1 1.2	560 FC 90 DY	560 FC 90 DY	ALL	ALL
	Lubricate the nose landing gear assembly. **INTERVAL NOTE:** Whichever comes first.						
32-085-00-01	MRB	32-21-71-200-803	1.1	30 MO	30 MO	ALL	ALL
	Inspect (detailed) nose landing gear axle.						
32-090-00-01	MRB	32-21-00-000-801 32-21-00-400-801 32-21-21-000-801 32-21-21-400-801 32-33-11-000-801 32-33-11-400-801 32-33-51-000-801 32-33-51-400-801	1.1 1.2	18000 FC 10 YR	18000 FC 10 YR	ALL	ALL
	Restore the nose landing gear assembly. **INTERVAL NOTE:** Whichever comes first.						
32-100-00-01	MRB	32-21-00-000-801 32-21-00-400-801	1.1	75000 FC	75000 FC	ALL	ALL
	Discard the nose landing gear life limited parts. **ACCESS NOTE:** Remove the aft access panel on the side wall of the nose wheel well.						
32-110-00-01	MRB	32-31-51-200-801	1.1	8000 FC	8000 FC	ALL	ALL
	Perform a detail visual inspection of the landing gear selector valve.						
32-120-00-01	MRB	32-21-00-200-801	1.1	6600 FC	6600 FC	ALL	ALL
	Perform a detail visual inspection of the nose landing gear extension and retraction mechanism. (Includes retract actuator, lock actuator, and lock mechanism)						
32-150-00-01	MRB	32-11-00-200-801	1.1	6600 FC	6600 FC	ALL	ALL
	Perform a detail visual inspection of the left main landing gear extension and retraction mechanism. (Includes retraction actuators, up lock actuators, down lock actuators, up lock mechanism, and down lock mechanism).						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
32-150-00-02	MRB	32-11-00-200-801	1.1	6600 FC	6600 FC	ALL	ALL
	Perform a detail visual inspection of the right main landing gear extension and retraction mechanism. (Includes retraction actuators, up lock actuators, down lock actuators, up lock mechanism, and down lock mechanism).						
32-200-00-01	MRB	32-35-00-730-801	1.1	36 MO	36 MO	ALL	ALL
	Perform a functional check of the nose landing gear manual extension system. In-flight check of the manual extension system under airline controlled operational procedures (non-revenue flight) is considered an optional method to accomplish this task.						
32-220-00-01	MRB	32-34-00-730-801	1.1	36 MO	36 MO	ALL	ALL
	Perform a functional check of the left main landing gear manual extension system and alternate extension bypass valve. In-flight check of the manual extension system under airline controlled operational procedures (non-revenue flight) is considered an optional method to accomplish this task.						
32-220-00-02	MRB	32-34-00-730-801	1.1	36 MO	36 MO	ALL	ALL
	Perform a functional check of the right main landing gear manual extension system and alternate extension bypass valve. In-flight check of the manual extension system under airline controlled operational procedures (non-revenue flight) is considered an optional method to accomplish this task.						
32-230-00-01	MRB	32-32-21-000-801 32-32-21-400-801	1.1	25000 FH	25000 FH	ALL	ALL
	Remove the left main landing gear wheel well protection retract pressure fuse for functional test.						
32-230-00-02	MRB	32-32-21-000-801 32-32-21-400-801	1.1	25000 FH	25000 FH	ALL	ALL
	Remove the right main landing gear wheel well protection retract pressure fuse for functional test.						
32-240-00-01	MRB	32-31-71-400-802	1.1	8000 FH	8000 FH	ALL	ALL
	Operationally check the landing gear transfer valve.						
32-250-00-01	MRB	32-41-81-000-802	1.1	6600 FC	6600 FC	ALL	ALL
	Perform a detail visual inspection of the forward and aft brake control linkages and cable quadrants.						
32-260-00-01	MRB	32-41-31-000-802	1.1	5000 FC	5000 FC	ALL	ALL
	Perform a detail visual inspection of the brake metering valves.						
32-270-01-01	MRB	32-41-41-700-801	1.1	50 FC	50 FC	ALL	ALL
	Visually check the left brake wear pins for minimum extension.						
32-270-02-01	MRB	32-41-41-700-801	1.1	50 FC	50 FC	ALL	ALL
	Visually check the right brake wear pins for minimum extension.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
32-290-00-01	MRB	32-41-00-710-803 32-41-93-700-801	1.1	15000 FC	15000 FC	ALL	ALL
	Perform an operational check of the alternate brake system and alternate brake selector valve.						
32-300-00-01	MRB	12-15-11-420-801 12-15-11-610-801	1.1	1200 FH	1200 FH	ALL	ALL
	Check brake accumulator precharge pressure, service as required.						
32-310-00-01	MRB	32-41-00-720-801	1.1	12500 FC	12500 FC	ALL	ALL
	Operationally check the brake accumulator isolation valve.						
32-330-00-01	MRB	32-44-11-000-805	1.1	8000 FC	8000 FC	ALL	ALL
	Perform a detail visual inspection of the parking brake mechanical control path.						
32-340-00-01	MRB	32-44-00-790-801	1.1	4000 FC	4000 FC	ALL	ALL
	Perform a functional bleed down check of the parking brake system and brake accumulator precharge pressure.						
32-350-00-01	MRB	12-15-51-610-802 12-15-51-780-801	1.1	48 HR	48 HR	ALL	ALL
	Check nose and main landing gear tires for proper inflation. Service as required. **INTERVAL NOTE:** 48 elapsed clock hours.						
32-360-00-01	MRB	32-45-00-700-801 32-45-00-700-803	1.1	48 HR	48 HR	ALL	ALL
	Visually check nose and main landing gear tires and wheels for condition and wear. **INTERVAL NOTE:** 48 Elapsed clock hours.						
32-380-00-01	MRB	32-45-21-000-801 32-45-21-400-801	1.1	NOTE		ALL	ALL
	Restore the nose landing gear wheel assemblies. **INTERVAL NOTE:** Tire change.						
32-380-01-01	MRB	32-45-11-000-801 32-45-11-400-801	1.1	NOTE		ALL	ALL
	Restore the left main landing gear wheel assemblies. **INTERVAL NOTE:** Tire change.						
32-380-02-01	MRB	32-45-11-000-801 32-45-11-400-801	1.1	NOTE		ALL	ALL
	Restore the right main landing gear wheel assemblies. **INTERVAL NOTE:** Tire change.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
32-390-00-01	MRB	32-51-00-200-801	1.1	8000 FC	8000 FC	ALL	ALL
	Perform a detail visual inspection of the nose wheel steering mechanical control path and rudder pedal steering interconnect control path including rotary actuator. **ACCESS NOTE:** Access 114AW is only applicable to airplanes with a right steering wheel.						
32-400-00-01	MRB	32-51-51-200-801	1.1	4000 FC	4000 FC	ALL	ALL
	Perform a detail visual inspection of the nose wheel steering actuator rod ends and associated hardware.						
32-420-00-01	MRB	32-09-10-710-801	1.1	25000 FH	25000 FH	ALL	ALL
	Perform an operational check (bite check) of the proximity switch electronics unit.						
32-430-00-01	MRB	32-71-00-200-801	1.1	300 FC	300 FC	800 900 900ER	ALL
	Perform a visual check of the tail skid crushable cartridge for evidence of a tail strike.						
32-440-00-01	MRB	20-20-31-200-801 20-20-31-200-802 20-20-31-200-805	1.1 1.2	4800 FC 24 MO	4800 FC 24 MO	ALL	ALL
	Perform a detail visual inspection of the control cables in the NLG wheel well for broken wires. Check associated pulleys and brackets for condition and security of installation. The following cables are located in the nose landing gear wheel well: A. NLG manual extension cables B. NLG steering cables Note: The control cable system must be displaced full travel in each direction for complete inspection at seals, pulleys, and fairlead areas. Observe that the manual extension mechanism release roller correctly resets to its retracted position. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** NLG covers P/N 273A4520-1 or 273A4520-2 need to be removed to perform this task.						
32-440-00-02	MRB	20-20-31-200-801 20-20-31-200-802	1.1 1.2	4800 FC 24 MO	4800 FC 24 MO	ALL	ALL
	Perform a detail visual inspection of the MLG manual extension cables in the left main landing gear wheel well for broken wires. Check associated pulleys, brackets, and mechanisms for condition and security of installation. Note: MLG manual extension cables do not require displacement for complete inspection. **INTERVAL NOTE:** Whichever comes first.						
32-440-00-03	MRB	20-20-31-200-801 20-20-31-200-802	1.1 1.2	4800 FC 24 MO	4800 FC 24 MO	ALL	ALL
	Perform a detail visual inspection of the MLG manual extension cables within the right main landing gear wheel well for broken wires. Check associated pulleys, brackets, and mechanisms for condition and security of installation. Note: MLG manual extension cables do not require displacement for complete inspection. **INTERVAL NOTE:** Whichever comes first.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
32-450-00-01	MRB	20-20-31-200-801 20-20-31-200-802 20-20-31-200-805	1.1 1.2	10400 FC 6 YR	10400 FC 6 YR	ALL	ALL

Perform a detail visual inspection of the control cables above the MLG wheel well for broken wires from B.S. 663.75 to B.S. 727. Check associated pulleys, brackets, and mechanisms for condition and security of installation. The following cables are located above the MLG wheel well:

A. Main landing gear control cables

B. MLG manual extension cables

C. MLG brake control cables

Note: The control cable system must be displaced full travel in each direction for complete inspection at seals, pulleys, and fairlead areas.

INTERVAL NOTE: Whichever occurs first.

ACCESS NOTE: Passenger cabin floor panels between B.S. 663.75 and B.S. 727

32-450-00-02	MRB	20-20-31-200-801 20-20-31-200-802 20-20-31-200-805 32-31-22-211-802	1.1 1.2	10400 FC 6 YR	10400 FC 6 YR	ALL	ALL

Perform a detail visual inspection of the control cables for broken wires within the lower nose compartment. Check associated pulleys, brackets, and mechanisms for condition and security of installation. The following cables are located within the lower nose compartment:

A. MLG control cables

B. MLG manual extension cables

C. MLG brake control cables

D. NLG Manual extension cables

E. NLG Steering cables

Note: The control cable system must be displaced full travel in each direction for complete inspection at seals, pulleys, and fairlead areas.

INTERVAL NOTE: Whichever occurs first.

32-450-00-03	MRB	20-20-31-200-801 20-20-31-200-802 20-20-31-200-805	1.1 1.2	10400 FC 6 YR	10400 FC 6 YR	ALL	ALL

Perform a detail visual inspection of the control cables for broken wires within the electronics compartment. Check associated pulleys, brackets, and mechanisms for condition and security of installation. The following cables are located in the electronics compartment:

A. MLG control cables

B. MLG manual extension cables

C. MLG brake control cables

Note: The control cable system must be displaced full travel in each direction for complete inspection at seals, pulleys, and fairlead areas.

INTERVAL NOTE: Whichever occurs first.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
32-450-00-04	MRB	20-20-31-200-801 20-20-31-200-802 20-20-31-200-805	1.1 1.2	10400 FC 6 YR	10400 FC 6 YR	ALL	ALL

Perform a detail visual inspection of the control cables for broken wires within the air conditioning distribution bay. Check associated pulleys, brackets, and mechanisms for condition and security of installation. The following cables are located within the air conditioning distribution bay:

A. MLG control cables

B. MLG manual extension cables

C. MLG brake control cables

Note: The control cable system must be displaced full travel in each direction for complete inspection at seals, pulleys, and fairlead areas.

INTERVAL NOTE: Whichever occurs first.

ACCESS NOTE: Forward cargo compartment aft bulkhead panels

32-450-00-05	MRB	20-20-31-200-801 20-20-31-200-802 20-20-31-200-805	1.1 1.2	10400 FC 6 YR	10400 FC 6 YR	ALL	ALL

Perform a detail visual inspection of the control cables within the forward cargo compartment for broken wires. Check associated pulleys, brackets, and mechanisms for condition and security of installation. The following cables are located within the forward cargo compartment:

A. MLG control cables

B. MLG manual extension cables

C. MLG brake control cables

Note: The control cable system must be displaced full travel in each direction for complete inspection at seals, pulleys, and fairlead areas.

INTERVAL NOTE: Whichever occurs first.

ACCESS NOTE: Forward Cargo Compartment Ceiling Panels or Floor Panels between B.S.396 to B.S. 540.

32-460-00-01	MRB	20-20-31-200-801 20-20-31-200-802 20-20-31-200-805 32-31-22-211-801	1.1 1.2	36000 FC 12 YR	36000 FC 12 YR	ALL	ALL

Perform a detail visual inspection of the control cables within the control cabin for broken wires. Check associated pulleys, brackets, and mechanisms for condition and security of installation. The following cables are located within the control cabin:

A. Landing gear control lever cables

B. Nose wheel steering cables

Note: The control cables must be displaced full travel in each direction for complete inspection at seals, pulleys, and fairlead areas.

INTERVAL NOTE: Whichever occurs first.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
32-460-00-02	MRB	20-20-31-200-801 20-20-31-200-802 20-20-31-200-805	1.1 1.2	36000 FC 12 YR	36000 FC 12 YR	ALL	ALL

Perform a detail visual inspection of the control cables for broken wires within the passenger compartment over the wing center section from B.S. 540 to B.S. 663.75. Check associated pulleys, brackets, and mechanisms for condition and security of installation. The following cables are located within the passenger compartment above the wing center section:

A. MLG control cables

B. MLG manual extension cables

C. MLG brake control cables

Note: The control cables must be displaced full travel in each direction for complete inspection at seals, pulleys, and fairlead areas.

INTERVAL NOTE: Whichever occurs first.

32-720-01-01	MRB	32-05-03-210-801 51-05-01-210-802	1.1	10 YR	10 YR	ALL	ALL

Inspect left main landing gear assembly, including outer cylinder, inner cylinder, axle, torsion links, side strut, lock links and retraction linkage. Landing gear removal is required. Disassemble as required to accomplish CPCP basic task on all fittings, lugs, lug bores, bolts and pins. Normal overhaul procedures, applied with the landing gear removed, at intervals not exceeding 10 years are adequate to maintain corrosion at safe levels on main landing gear components. Therefore application of the basic tasks and reporting are not required on these components.

32-720-02-01	MRB	32-05-03-210-802 51-05-01-210-802	1.1	10 YR	10 YR	ALL	ALL

Inspect right main landing gear assembly, including outer cylinder, inner cylinder, axle, torsion links, side strut, lock links and retraction linkage. Landing gear removal is required. Disassemble as required to accomplish CPCP basic task on all fittings, lugs, lug bores, bolts and pins. Normal overhaul procedures, applied with the landing gear removed, at intervals not exceeding 10 years, are adequate to maintain corrosion at safe levels on main landing gear components. Therefore application of the basic tasks and reporting are not required on these components.

32-750-00-01	MRB	32-05-03-210-803 51-05-01-210-802	1.1	10 YR	10 YR	ALL	ALL

Inspect nose landing gear assembly, including outer cylinder, inner cylinder, drag strut, lock links, torsion links, and steering mechanism (plates and collar). Landing gear removal is required. Disassemble as required to accomplish CPCP basic task on all fittings, lugs, lug bores, bolts and pins. Normal overhaul procedures, applied with the landing gear removed, at intervals not exceeding 10 years, are adequate to maintain corrosion at safe levels on nose landing gear components. Therefore application of the basic tasks and reporting are not required on these components.

32-800-00-01	MRB	05-41-07-210-801	1.1	120 DY	120 DY	ALL	ALL

Perform an external zonal inspection (GV) of the nose landing gear and landing gear doors. Inspection is accomplished from the ground, without the use of stands or ladders. (EZAP)

INTERVAL NOTE: The EZAP inspection requirement with interval 5500 FC/30 MO is satisfied by this zonal inspection.

32-804-01-01	MRB	05-41-07-210-802	1.1 1.2	120 DY 745 FC	120 DY 745 FC	ALL	ALL

Perform an external zonal inspection (GV) of the left main landing gear and landing gear doors. Inspection is accomplished from the ground, without the use of stands or ladders.

INTERVAL NOTE: Whichever comes first.

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
32-806-01-01	MRB	05-41-07-210-803	1.1 1.2	5500 FC 30 MO	5500 FC 30 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the left main landing gear and landing gear doors. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 5500 FC/30 MO is satisfied by this zonal inspection.						
32-808-02-01	MRB	05-41-07-210-804	1.1 1.2	120 DY 745 FC	120 DY 745 FC	ALL	ALL
	Perform an external zonal inspection (GV) of the right main landing gear and landing gear doors. Inspection is accomplished from the ground, without the use of stands or ladders. **INTERVAL NOTE:** Whichever comes first.						
32-810-02-01	MRB	05-41-07-210-805	1.1 1.2	5500 FC 30 MO	5500 FC 30 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the right main landing gear and landing gear doors. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 5500 FC/30 MO is satisfied by this zonal inspection.						
33-010-00-01	MRB	33-51-00-710-801	1.1	600 FH	600 FH	ALL	ALL
	Operational check of the emergency lights.						
33-020-00-01	MRB	33-51-00-720-801	1.1	2 YR	2 YR	ALL	ALL
	Operational check of the flight deck emergency lighting switch "on" and "armed" position and attendant panel emergency light switch "on" position.						
33-055-00-01	MRB	33-51-06-200-801	1.1	1 YR	1 YR	ALL	ALL
	Functionally check the emergency lights battery packs for capacity (15 min. minimum) and one complete deep cycle.						
33-060-00-01	MRB	33-51-06-600-802	1.1	2 YR	2 YR	ALL	ALL
	Restore (two or more complete deep cycles) battery capacity to required standard. Note: The second deep cycle is accomplished by MPD item 33-055-00.						
33-070-00-01	MRB	33-51-15-860-802	1.1 1.2	3 YR 12000 FH	3 YR 12000 FH	ALL	ALL
	Functional check of the photoluminescent floor proximity lighting. **AIRPLANE NOTE:** If installed. **INTERVAL NOTE:** Whichever occurs first.						
33-080-00-01	MRB	33-51-15-960-801	1.1	10 YR	10 YR	ALL	ALL
	Replace photoluminescent floor proximity lighting at manufacturer's life limit. **AIRPLANE NOTE:** If installed.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
33-090-00-01	MPD	57-21-22-200-801	1.1	6 MO	6 MO	ALL	ALL
	Perform a general visual inspection of the forward position light lens on the winglet. Note: This task satisfies the requirement of the Airplane Partners Boeing (APB) task 33-W01-00. **AIRPLANE NOTE:** Single lens configuration only.						
33-090-01-01	MPD	57-21-22-200-802	1.1	24 MO	24 MO	ALL	ALL
	Perform a general visual inspection of the forward position light lens on the winglet - Dual Lens Configuration (Glass). Note: This task satisfies the requirement of the Airplane Partners Boeing (APB) task 33-W02-00. **AIRPLANE NOTE:** Dual Lens Configuration (Glass).						
34-010-00-01	MRB	34-11-00-790-810	1.1	24 MO	24 MO	ALL	ALL
	Functional leak check of captain's pitot system.						
34-020-00-01	MRB	34-11-00-790-811	1.1	24 MO	24 MO	ALL	ALL
	Functional leak check of first officer's pitot system.						
34-030-00-01	MRB	34-11-00-790-812	1.1	24 MO	24 MO	ALL	ALL
	Functional leak check of standby pitot system.						
34-040-00-01	MRB	34-11-00-790-804	1.1	24 MO	24 MO	ALL	ALL
	Functional leak check of captain's static system.						
34-050-00-01	MRB	34-11-00-790-806	1.1	24 MO	24 MO	ALL	ALL
	Functional leak check of first officer's static system.						
34-060-00-01	MRB	34-11-00-790-808	1.1	24 MO	24 MO	ALL	ALL
	Functional leak check of standby static system.						
34-060-10-01	MPD	34-11-00-780-802	1.1	72 MO	72 MO	ALL	ALL
	Functional check of the air data system altimetry system.						
34-070-00-01	MRB	34-11-01-200-804	1.1	7500 FH	7500 FH	ALL	ALL
	Detail visual inspection of the pitot probes.						
34-080-00-01	MRB	34-11-02-200-803	1.1	15000 FH	15000 FH	ALL	ALL
	Detailed inspection of the static ports.						
34-090-00-01	MRB	34-11-00-210-801	1.1	15000 FH	15000 FH	600 700 800 900 900ER	ALL
	Detail inspection for moisture in the pitot systems.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
34-100-00-01	MRB	34-11-00-210-802	1.1	15000 FH	15000 FH	600 700 800 900 900ER	ALL
	Detail inspection for moisture in the static systems.						
34-110-00-01	MRB	34-53-00-730-802	1.1	24 MO	24 MO	ALL	ALL
	Perform a functional check of the ATC TRANSPONDER SYSTEM - Test Procedure for airplanes WITHOUT Elementary Surveillance (ELS), Enhanced Surveillance (EHS), Extended Squitter (ES) features using the T-48 or T-49 Test Set. (See reference FAR 91.413) **INTERVAL NOTE:** or national requirement.						
34-110-00-02	MRB	34-53-00-730-803 34-53-00-730-805 34-53-00-730-806	1.1	24 MO	24 MO	ALL	ALL
	Perform a functional check of the ATC TRANSPONDER SYSTEM - Test Procedure for airplanes with or without Elementary Surveillance (ELS), Enhanced Surveillance (EHS), Extended Squitter (ES) features using either the IFR ATC-601, IFR 6000 or TR220 Test Set. (See reference FAR 91.413) **INTERVAL NOTE:** or national requirement.						
34-130-00-01	MRB	34-24-03-000-801 34-24-03-400-801	1.1	3 YR	3 YR	ALL	ALL
	Discard the dedicated battery/charger internal battery for the integrated standby flight display. **AIRPLANE NOTE:** If Installed. **INTERVAL NOTE:** At manufacture's life limit.						
34-140-00-01	MRB	34-24-02-710-802	1.1	9000 FH	9000 FH	ALL	ALL
	Operationally check the Integrated Standby Flight Display Dedicated Battery/Charger. **AIRPLANE NOTE:** If Installed.						
35-010-00-01	MRB	35-12-00-700-802	1.1	6000 FH	6000 FH	ALL	ALL
	Operationally check each flight crew oxygen mask/regulator (out of the box assembly).						
35-020-00-01	MRB	35-12-85-000-802 35-12-85-400-802	1.1	16000 FH	16000 FH	ALL	ALL
	Functionally check (off the airplane) each flight crew oxygen mask/regulator per the manufacturer's component manual.						
35-040-00-02	MRB	12-15-21-600-803-002	1.1	VEN REC		ALL	ALL
	Discard the flight crew oxygen cylinder. **INTERVAL NOTE:** At Vendors recommendation.						
35-050-00-01	MRB	35-12-00-710-801	1.1	2400 FH	2400 FH	ALL	ALL
	Visually (cross) check the flight crew oxygen cylinder pressure indicator and the control compartment flight crew oxygen indicator.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
35-060-00-01	MRB	35-22-00-210-801	1.1	6 YR	6 YR	ALL	ALL
	Visually check the temperature sensitive tape on each passenger cabin, (including lavatory and cabin attendant's), chemical oxygen generator. **AIRPLANE NOTE:** Not applicable to airplanes with all gaseous passenger oxygen system.						
35-065-00-01	MRB	35-22-00-210-802	1.1	6 YR	6 YR	ALL	ALL
	Visual check of the passenger lavatory oxygen Constant Dispensing System (CDS) discharge indicator tape for evidence of rupture and verify oxygen activation pin has not been pulled out. **AIRPLANE NOTE:** If lavatory oxygen Constant Dispensing System (CDS) installed.						
35-070-00-01	MRB	35-22-11-000-804-001 35-22-11-000-811-001 35-22-11-400-804-001 35-22-11-400-811-001	1.1	NOTE		ALL	ALL
	Discard the chemical oxygen generators. **AIRPLANE NOTE:** Not applicable to airplanes with all gaseous passenger oxygen system. **INTERVAL NOTE:** At Vendors recommendation.						
35-070-00-03	MRB	35-22-11-000-805-001 35-22-11-000-811-001 35-22-11-400-805-001 35-22-11-400-811-001	1.1	NOTE		ALL	ALL
	Discard the chemical oxygen generators. **AIRPLANE NOTE:** Not applicable to airplanes with all gaseous passenger oxygen system. **INTERVAL NOTE:** At Vendors recommendation.						
35-070-00-04	MRB	35-22-11-000-806-001 35-22-11-000-811-001 35-22-11-400-806-001 35-22-11-400-811-001	1.1	NOTE		ALL	ALL
	Discard the chemical oxygen generators. **AIRPLANE NOTE:** Not applicable to airplanes with all gaseous passenger oxygen system. **INTERVAL NOTE:** At Vendors recommendation.						
35-075-00-01	MRB	35-22-51-000-801 35-22-51-040-801 35-22-51-400-801 35-22-51-440-801	1.1	LIF LIM		ALL	ALL
	Discard the passenger lavatory Constant Dispensing System (CDS) oxygen cylinder. **AIRPLANE NOTE:** If lavatory oxygen Constant Dispensing System (CDS) installed. **INTERVAL NOTE:** Vendor recommended life limit.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
35-080-00-01	MRB	35-22-00-700-801 35-22-00-700-802	1.1	15000 FH	15000 FH	ALL	ALL
	Functionally check the passenger oxygen system automatic and manual modes, (this check includes the PSU/ASU door latch actuator/solenoids and oxygen system altitude pressure switch(es), and for gaseous systems this also includes the voltage averaging unit, cylinder pressure transducer and flight deck pressure indication).						
35-090-00-01	MRB	35-22-31-210-801-001	1.1	12000 FH	12000 FH	ALL	ALL
	Detailed visual inspection 10% (rotational inspection) of the passenger, lavatory, and attendant oxygen masks for condition and security.						
35-100-00-01	MRB	35-31-00-710-801	1.1	8000 FH	8000 FH	ALL	ALL
	Visually check all the portable oxygen cylinders for presence, condition, and security.						
35-120-00-01	MRB	35-31-01-960-801	1.1	NOTE		ALL	ALL
	Discard the portable oxygen cylinder. **INTERVAL NOTE:** At Vendors recommendation.						
36-020-01-01	MRB	36-12-00-710-802	1.1	16000 FH	16000 FH	ALL	ALL
	Functionally check the left precooler control valve and wing TAI solenoid.						
36-020-02-01	MRB	36-12-00-710-802	1.1	16000 FH	16000 FH	ALL	ALL
	Functionally check the right precooler control valve and wing TAI solenoid.						
36-030-01-01	MRB	36-12-03-000-801 36-12-03-400-801	1.1	16000 FH	16000 FH	ALL	ALL
	Functionally check (off-airplane) the left precooler control valve sensor per vendor's overhaul manual. Task card procedures apply to on-airplane portion only (removal/installation).						
36-030-02-01	MRB	36-12-03-000-801 36-12-03-400-801	1.1	16000 FH	16000 FH	ALL	ALL
	Functionally check (off-airplane) the right precooler control valve sensor per vendor's overhaul manual. Task card procedures apply to on-airplane portion only (removal/installation).						
38-030-00-01	MRB	38-32-05-960-801	1.1	2500 FH	2500 FH	ALL	ALL
	Restore (clean) or replace the vacuum blower filter (if installed). **AIRPLANE NOTE:** Applicable to airplanes with vacuum blower equipped with a filter.						
38-040-00-01	MRB	38-32-03-000-801 38-32-03-400-801	1.1	60 MO	60 MO	ALL	ALL
	Restore the waste drain ball-valve by replacing the seals (2) (off aircraft). Note: The intent of the task is to restore the ball valve assembly by replacing two internal seals (P/N RS823-1). The seals are located above and below the ball valve inside the valve assembly.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
38-060-00-01	MRB	38-42-10-960-801	1.1	6500 FH	6500 FH	ALL	ALL
	Discard and replace the bleed air in-line filter.						
38-090-00-01	MRB	38-32-02-000-801 38-32-02-400-801	1.1	2500 FH	2500 FH	ALL	ALL
	Restore (clean or replace) the waste tank water separator filter baskets. **AIRPLANE NOTE:** Applicable to filter basket part numbers 01940-001 and 01956-000.						
38-100-00-01	MRB	38-32-02-000-801 38-32-02-100-801 38-32-02-400-801	1.1	6000 FH	6000 FH	ALL	ALL
	Restore (clean or replace) the waste tank water separator filter baskets. **AIRPLANE NOTE:** Applicable to all filter basket part numbers except 01940-001 and 01956-000.						
47-200-00-02	MSG3	47-21-00-700-802	1.1	6500 FH	6500 FH	ALL	ALL
	Visual check fluid accumulation in the nitrogen generation system (NGS) tubing through the drain cap. **AIRPLANE NOTE:** If Nitrogen Generation System is installed.						
47-210-00-01	MSG3	47-00-00-710-801	1.1	13000 FH	13000 FH	ALL	ALL
	Operationally check center tank Vent Cross Flow Check Valve. **SPECIAL NOTE:** AWL task (47-AWL-06) interval for this task is 13000 FH. See MPD Section 9. **AIRPLANE NOTE:** If Nitrogen Generation System is installed.						
47-220-00-01	MSG3	47-00-00-790-802	1.1	6500 FH	6500 FH	ALL	ALL
	Inspect (detailed) the nitrogen enriched air (NEA) distribution lines from the air separation module (ASM) to the fuel tank rear spar for damage and leaks. **SPECIAL NOTE:** AWL task (47-AWL-07) interval for this task is 6500 FH. See MPD Section 9. **AIRPLANE NOTE:** If Nitrogen Generation System is installed.						
47-300-00-01	MSG3	47-32-02-000-801 47-32-02-400-801	1.1	12000 FH	12000 FH	ALL	ALL
	Restore the ozone converter (off-aircraft). **AIRPLANE NOTE:** If Nitrogen Generation System is installed.						
47-310-00-01	MSG3	47-32-03-000-801 47-32-03-400-801	1.1	12000 FH	12000 FH	ALL	ALL
	Clean the nitrogen generation system heat exchanger (off aircraft). **AIRPLANE NOTE:** If Nitrogen Generation System is installed.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
47-400-00-01	MSG3	47-43-02-720-801	1.1	22500 FH	22500 FH	ALL	ALL
	Functional test of the thermal switch (off aircraft). **SPECIAL NOTE:** AWL task (47-AWL-04) interval for this task is 22500 FH. See MPD Section 9. **AIRPLANE NOTE:** If Nitrogen Generation System is installed.						
49-010-00-01	MRB	49-13-11-200-803	1.1	5 YR	5 YR	ALL	ALL
	Perform a general visual inspection of the APU mounts for general condition and security of installation.						
49-020-00-01	MRB	49-13-11-200-801	1.1	8 YR	8 YR	ALL	ALL
	Perform a detailed inspection of the APU mounts.						
49-030-00-01	MRB	49-15-11-200-801	1.1	APU CNG		ALL	ALL
	Perform a detailed inspection of the sigma seal (after APU removal).						
49-040-00-01	MRB	49-17-11-200-801	1.1	APU CNG		ALL	ALL
	Perform a detailed inspection of the APU insulation panels. (After APU removal).						
49-052-00-01	MRB	49-11-00-000-801 49-11-00-400-801	1.1	LIF LIM		ALL	ALL
	Discard the engine compressor impeller. **INTERVAL NOTE:** Refer to APU shop manual for life limits.						
49-062-00-01	MRB	49-11-00-000-801 49-11-00-400-801	1.1	LIF LIM		ALL	ALL
	Discard the first stage turbine disk. **INTERVAL NOTE:** Refer to APU shop manual for life limits.						
49-072-00-01	MRB	49-11-00-000-801 49-11-00-400-801	1.1	LIF LIM		ALL	ALL
	Discard the second stage turbine rotor. **INTERVAL NOTE:** Refer to APU shop manual for life limits.						
49-082-00-01	MRB	49-11-00-000-801 49-11-00-400-801	1.1	LIF LIM		ALL	ALL
	Discard the turbine shaft. **INTERVAL NOTE:** Refer to APU shop manual for life limits.						
49-102-00-01	MRB	49-31-21-000-801 49-31-21-400-801	1.1	4000 AH	4000 AH	ALL	ALL
	Discard the fuel inlet filter element on the fuel control unit (FCU).						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
49-140-00-01	MRB	49-61-00-710-801	1.1	1600 AH	1600 AH	ALL	ALL
	Perform an operational check of the following by interrogating the CDU's APU maintenance pages. -APU data memory module (DMM) -Electronics control unit (ECU) -Speed Sensor -EGT rake						
49-172-00-01	MRB	49-81-11-200-801	1.1	19000 AH	19000 AH	ALL	ALL
	Inspect (detailed) the APU exhaust seal.						
49-212-00-01	MRB	49-81-41-200-801	1.1	10000 AH	10000 AH	ALL	ALL
	Perform a general visual inspection of the eductor (on the APU) for general condition.						
49-220-00-01	MRB	49-91-71-200-801	1.1	25000 FH	25000 FH	ALL	ALL
	Inspect (detailed) the eductor inlet duct (interior and exterior).						
49-240-00-01	MRB	49-15-22-600-801	1.1	16000 FH	16000 FH	ALL	ALL
	Lubricate the vortex generator hinge pin.						
52-010-00-01	MRB	12-25-11-640-802	1.1	2 YR	2 YR	ALL	ALL
	Lubricate the forward entry door handle, latch mechanisms (latch torque tube bearings and latch control rods), the bearings on the door hinge torque tube and the fwd entry door control rods and stop rods.						
52-010-00-02	MRB	12-25-13-640-802	1.1	2 YR	2 YR	ALL	ALL
	Lubricate the forward service door handle, latch mechanisms (latch torque tube bearings and latch control rods) and the bearings on the door hinge torque tube.						
52-010-00-03	MRB	12-25-12-640-802	1.1	2 YR	2 YR	ALL	ALL
	Lubricate the aft entry door handle, latch mechanisms (latch torque tube bearings and latch control rods) and the bearings on the door hinge torque tube.						
52-010-00-04	MRB	12-25-13-640-802	1.1	2 YR	2 YR	ALL	ALL
	Lubricate the aft service door handle, latch mechanisms (latch torque tube bearings and latch control rods) and the bearings on the door hinge torque tube.						
52-020-00-01	MRB	12-25-11-640-801	1.1	1 YR	1 YR	ALL	ALL
	Lubricate the forward entry door guide plate tracks and arm assemblies (rod end bearings and threads), the upper and lower hinge arm bushing, gate hinges and the fuselage hinge torque tube bearing.						
52-020-00-02	MRB	12-25-13-640-801	1.1	1 YR	1 YR	ALL	ALL
	Lubricate the forward service door guide plate tracks and arm assemblies (rod end bearings and threads), the upper and lower hinge arm bushings and gate hinges.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
52-020-00-03	MRB	12-25-12-640-801	1.1	1 YR	1 YR	ALL	ALL
	Lubricate the aft entry door guide plate tracks and arm assemblies (rod end bearings and threads), the upper and lower hinge arm bushing and gate hinges.						
52-020-00-04	MRB	12-25-13-640-801	1.1	1 YR	1 YR	ALL	ALL
	Lubricate the aft service door guide plate tracks and arm assemblies (rod end bearings and threads), the upper and lower hinge arm bushings and gate hinges.						
52-030-00-01	MRB	52-11-00-200-802	1.1	3 YR	3 YR	ALL	ALL
	Inspect (detailed) the forward entry door centering guide stud and nylon track pads for condition.						
52-040-00-01	MRB	52-41-00-200-803	1.1	3 YR	3 YR	ALL	ALL
	Inspect (Detailed) the fwd service door centering guide bearings for condition.						
52-040-00-02	MRB	52-13-00-200-803	1.1	3 YR	3 YR	ALL	ALL
	Inspect (Detailed) the aft entry door centering guide bearings for condition.						
52-040-00-03	MRB	52-41-00-200-803	1.1	3 YR	3 YR	ALL	ALL
	Inspect (Detailed) the aft service door centering guide bearing for condition.						
52-050-00-01	MRB	52-11-00-200-803 52-11-00-200-804	1.1	6000 FH	6000 FH	ALL	ALL
	Inspect (General Visual) the forward entry door pressure and flapper seals for degradtion.						
52-050-00-02	MRB	52-41-00-200-802	1.1	6000 FH	6000 FH	ALL	ALL
	Inspect (General Visual) the forward service door pressure seal for degradation.						
52-050-00-03	MRB	52-13-00-200-802	1.1	6000 FH	6000 FH	ALL	ALL
	Inspect (General Visual) the aft entry door pressure seal for degradation.						
52-050-00-04	MRB	52-41-00-200-802	1.1	6000 FH	6000 FH	ALL	ALL
	Inspect (General Visual) the aft service door pressure seal for degradation.						
52-090-00-01	MRB	12-25-31-640-801	1.1	1 YR	1 YR	ALL	ALL
	Lubricate the forward cargo compartment door latch torque tube bearings and the counter balance idler crank.						
52-090-00-02	MRB	12-25-31-640-801	1.1	1 YR	1 YR	ALL	ALL
	Lubricate the aft cargo compartment door latch torque tube bearings and the counter balance idler crank.						
52-100-00-01	MRB	52-31-00-200-802	1.1	5000 FH	5000 FH	ALL	ALL
	Inspect (General Visual) the forward cargo compartment door pressure seal for degradation.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
52-100-00-02	MRB	52-31-00-200-802	1.1	5000 FH	5000 FH	ALL	ALL
	Inspect (General Visual) the aft cargo compartment door pressure seal for degradation.						
52-120-00-01	MRB	12-25-41-640-801	1.1	2 YR	2 YR	ALL	ALL
	Lubricate the E/E access door handle latching mechanism (rack and pinion gear and the lock pins).						
52-130-00-01	MRB	52-48-41-200-802	1.1	8000 FH	8000 FH	ALL	ALL
	Inspect (general visual) the E/E access door pressure seal for degradation.						
52-140-00-01	MRB	52-48-31-200-802	1.1	15000 FH	15000 FH	ALL	ALL
	Inspect (general visual) the forward access door pressure seal for degradation.						
52-200-00-01	MRB	52-71-11-710-801 52-71-22-710-803 52-71-31-710-801 52-71-41-710-801 52-71-42-710-801	1.1	5000 FH	5000 FH	ALL	ALL
	Operationally check the door sensors (proximity or mechanical switches as applicable) for the passenger cabin entry/service, E/E access, automatic overwing emergency exit, forward access and cargo doors. **ACCESS NOTE:** Access panels 832 and 842 are applicable to 737-800 and 737-900 only.						
52-210-00-01	MRB	52-22-00-710-802	1.1	15000 FH	15000 FH	ALL	ALL
	Operationally check the flight lock mechanical switches for the automatic overwing emergency exit doors. **ACCESS NOTE:** Zones and access panels 832 and 842 are applicable to 737-800 and 737-900 only.						
52-220-00-01	MRB	52-22-00-710-801	1.1	9 YR	9 YR	ALL	ALL
	Operationally check (cycle) the automatic overwing emergency exit doors. **ACCESS NOTE:** Zones and access panels 832 and 842 are applicable to 737-800 and 737-900 only.						
52-230-00-01	MRB	52-22-00-210-801	1.1	6 YR	6 YR	ALL	ALL
	Perform a detailed visual inspection of the automatic overwing emergency exit door latch rollers, links and pivot fittings/joints. **ACCESS NOTE:** Zones and access panels 832 and 842 are applicable to 737-800 and 737-900 only.						
52-240-00-01	MRB	52-22-00-710-803	1.1	6 YR	6 YR	ALL	ALL
	Operationally check the flight lock engagement and disengagement. **ACCESS NOTE:** Zones and access panels 832 and 842 are applicable to 737-800 and 737-900 only.						
52-250-00-01	MRB	52-22-00-210-802	1.1	6 YR	6 YR	ALL	ALL
	Perform a detailed visual inspection of automatic overwing exit door flight locks for corrosion and condition. **ACCESS NOTE:** Zones and access panels 832 and 842 are applicable to 737-800 and 737-900 only.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
52-360-00-01	MRB	52-51-00-710-802	1.1	30000 FH	30000 FH	ALL	ALL
	Perform a functional check of the locking and unlocking latch bolt mechanism on the flight deck door decompression panel. **SPECIAL NOTE:** CMR task (52-CMR-01) interval for this task is 3000 FH. See MPD Section 9. **AIRPLANE NOTE:** Applicable to airplane L/N 1221 and on and to those airplanes with the new flight deck security door installed by the customer specific Boeing service bulletins. **INTERVAL NOTE:** The equivalent CMR task (52-CMR-01) is performed at 3000 hours, which has precedence over the MRB interval of 30000 hours.						
52-370-00-01	MRB	52-51-00-710-803	1.1	30000 FH	30000 FH	ALL	ALL
	Operationally check the flight deck door decompression panel hinges. **AIRPLANE NOTE:** Applicable to airplane L/N 1221 and on and to those airplanes with the new flight deck security door installed by the customer specific Boeing service bulletins.						
52-380-00-01	MRB	52-51-00-210-801	1.1	30000 FH	30000 FH	ALL	ALL
	General visual inspection (GVI) of the flight deck door decompression panel hinges for condition and security. **AIRPLANE NOTE:** Applicable to airplane L/N 1221 and on and to those airplanes with the new flight deck security door installed by the customer specific Boeing service bulletins.						
52-390-00-01	MRB	52-51-01-200-801	1.1	6000 FH	6000 FH	ALL	ALL
	General visual inspection (GVI) of the flight deck door seals for condition and security. **AIRPLANE NOTE:** Applicable to airplane L/N 1221 and on and to those airplanes with the new flight deck security door installed by the customer specific Boeing service bulletins.						
52-400-00-01	MRB	52-51-00-210-802	1.1	30000 FH	30000 FH	ALL	ALL
	General visual inspection of the flight deck door decompression panel seals for condition and security. **AIRPLANE NOTE:** Applicable to airplane L/N 1221 and on and to those airplanes with the new flight deck security door installed by the customer specific Boeing service bulletins.						
52-410-00-01	MRB	52-51-00-700-802	1.1	11000 FH	11000 FH	ALL	ALL
	Functionally check the "deny" time delay function of the flight deck security door access system to verify; operation of the three position rotary switch in the P8 panel, the deny function, and reversion to the default mode. **AIRPLANE NOTE:** Applicable to airplane L/N 1221 and on and to those airplanes with the new flight deck security door installed by the customer specific Boeing service bulletins.						
52-450-00-01	MRB	51-05-01-210-803 52-05-03-211-802	1.1	9 YR	9 YR	ALL	ALL
	Inspect Flight Deck Security Door Assembly, including Main Door Panel Assy, Main Door Panel Bond Assy, and Armor Laminate Assy. **AIRPLANE NOTE:** Applicable to airplane L/N 1221 and on and to those airplanes with the New Flight Deck Security Door installed by the customer specific Boeing Service Bulletins. **ACCESS NOTE:** Disassemble door only if evidence of damage, fatigue, delamination, and or bulging is found.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
52-460-00-01	MRB	51-05-01-210-803 52-05-03-210-803	1.1	9 YR	9 YR	ALL	ALL

Inspect Flight Deck Security Door Surround Assembly, including Header Assembly, Post Assembly (right hand post), Latch and Deadbolt Receiver Assembly, Support Structure, Post Cover Armor Assembly and Hinge Assembly.

AIRPLANE NOTE: Applicable to airplane L/N 1221 and on and to those airplanes with the New Flight Deck Security Door installed by the customer specific Boeing Service Bulletins.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: As visible with carpet, tapestries (if equipped) and kick strips displaced.

52-470-00-01	MRB	51-05-01-210-803 52-05-03-211-803	1.1	9 YR	9 YR	ALL	ALL

Inspect Flight Deck Security Door Surround Assembly, including Header Assembly, Post Assembly (right hand post), Latch and Deadbolt Receiver Assembly, Support Structure, Post Cover Armor Assembly and Hinge Assembly.

AIRPLANE NOTE: Applicable to airplane L/N 1221 and on and to those airplanes with the New Flight Deck Security Door installed by the customer specific Boeing Service Bulletins.

ACCESS NOTE: For access displace interior furnishings including closets, lavs, galleys (if equipped) adjacent to door frames, and ceiling panels above door.

52-490-00-01	MRB	51-05-01-210-809 52-05-03-211-828	1.1	9 YR	9 YR	ALL	ALL

Inspect the flight deck door latch and hinge support assemblies.

AIRPLANE NOTE: Applicable to airplane L/N 1221 and on and to those airplanes with the New Flight Deck Security Door installed by the customer specific Boeing Service Bulletins.

ACCESS NOTE: Disassemble door only if evidence of damage, fatigue, delamination, and or bulging is found.

52-510-00-01	MRB	51-05-01-210-809 52-05-03-211-805	1.1 1.2	36 MO 6600 FC	36 MO 6600 FC	ALL	ALL

Inspect forward access door stop fittings and pins.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Inspect with door opened and lining not removed.

52-530-00-01	MRB	51-05-01-210-804 52-05-03-210-806	1.1 1.2	9 YR 18000 FC	8 YR 18000 FC	ALL	ALL

Inspect forward access door skin and structure.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Inspect with door removed.

52-540-00-01	MRB	51-05-01-210-809 52-05-03-211-806	1.1 1.2	36 MO 4000 FC	36 MO 4000 FC	ALL	ALL

Inspect E/E equipment compartment access door stop fittings and pins.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Inspect with door removed as required.

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
52-550-00-01	MRB	51-05-01-210-809 52-05-03-211-807	1.1 1.2	8 YR 18000 FC	8 YR 18000 FC	ALL	ALL
	Inspect E/E equipment compartment access door stop fittings and pins. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Inspect with door and access panel removed. Remove dagger pins as required.						
52-570-00-01	MRB	51-05-01-210-804 52-05-03-210-807	1.1 1.2	9 YR 18000 FC	8 YR 18000 FC	ALL	ALL
	Inspect E/E equipment compartment door skin and structure. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Inspect with door and access panel removed. Remove dagger pins as required.						
52-610-00-01	MRB	51-05-01-210-809 52-05-03-211-809	1.1 1.2	36 MO 4000 FC	36 MO 4000 FC	ALL	ALL
	Inspect forward entry door stop fittings and pins. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Inspect with doors opened and lining not removed.						
52-610-00-02	MRB	51-05-01-210-809 52-05-03-211-810	1.1 1.2	36 MO 4000 FC	36 MO 4000 FC	ALL	ALL
	Inspect forward galley service door stop fittings and pins. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Inspect with doors opened and lining not removed.						
52-610-00-03	MRB	51-05-01-210-809 52-05-03-211-811	1.1 1.2	36 MO 4000 FC	36 MO 4000 FC	ALL	ALL
	Inspect aft entry door stop fittings and pins. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Inspect with doors opened and lining not removed.						
52-610-00-04	MRB	51-05-01-210-809 52-05-03-211-812	1.1 1.2	36 MO 4000 FC	36 MO 4000 FC	ALL	ALL
	Inspect aft galley service door stop fittings and pins. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Inspect with doors opened and lining not removed.						
52-620-00-01	MRB	51-05-01-210-810 52-05-03-211-813	1.1 1.2	9 YR 18000 FC	8 YR 18000 FC	ALL	ALL
	Inspect forward entry door stop fittings and pins. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Remove insulation, interior liners and access panels as required.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
52-620-00-02	MRB	51-05-01-210-810 52-05-03-211-814	1.1 1.2	9 YR 18000 FC	8 YR 18000 FC	ALL	ALL
	Inspect forward galley service door stop fittings and pins. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Remove insulation, interior liners and access panels as required.						
52-620-00-03	MRB	51-05-01-210-810 52-05-03-211-815	1.1 1.2	9 YR 18000 FC	8 YR 18000 FC	ALL	ALL
	Inspect aft entry door stop fittings and pins. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Remove insulation, interior liners and access panels as required.						
52-620-00-04	MRB	51-05-01-210-810 52-05-03-211-816	1.1 1.2	9 YR 18000 FC	8 YR 18000 FC	ALL	ALL
	Inspect aft galley service door stop fittings and pins. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Remove insulation, interior liners and access panels as required.						
52-650-00-01	MRB	51-05-01-210-808 52-05-03-210-810	1.1 1.2	9 YR 18000 FC	8 YR 18000 FC	ALL	ALL
	Inspect forward entry door skin and structure. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Remove insulation, interior liners and access panels as required.						
52-650-00-02	MRB	51-05-01-210-808 52-05-03-210-811	1.1 1.2	9 YR 18000 FC	8 YR 18000 FC	ALL	ALL
	Inspect forward galley service door skin and structure. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Remove insulation, interior liners and access panels as required.						
52-650-00-03	MRB	51-05-01-210-808 52-05-03-210-812	1.1 1.2	9 YR 18000 FC	8 YR 18000 FC	ALL	ALL
	Inspect aft entry door skin and structure. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Remove insulation, interior liners and access panels as required.						
52-650-00-04	MRB	51-05-01-210-808 52-05-03-210-813	1.1 1.2	9 YR 18000 FC	8 YR 18000 FC	ALL	ALL
	Inspect aft galley service door skin and structure. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Remove insulation, interior liners and access panels as required.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
52-670-00-01	MRB	51-05-01-210-809 52-05-03-211-817	1.1 1.2	36 MO 6600 FC	36 MO 6600 FC	ALL	ALL
	Inspect the forward cargo door stop fittings and pins. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Inspect with doors opened and lining not removed.						
52-670-00-02	MRB	51-05-01-210-809 52-05-03-211-818	1.1 1.2	36 MO 6600 FC	36 MO 6600 FC	ALL	ALL
	Inspect the aft cargo door stop fittings and pins. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Inspect with doors opened and lining not removed.						
52-680-00-01	MRB	51-05-01-210-811 52-05-03-211-819	1.1 1.2	9 YR 18000 FC	8 YR 18000 FC	ALL	ALL
	Inspect forward cargo door stop fittings and pins. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Inspect with insulation blanket removed.						
52-680-00-02	MRB	51-05-01-210-811 52-05-03-211-820	1.1 1.2	9 YR 18000 FC	8 YR 18000 FC	ALL	ALL
	Inspect aft cargo door stop fittings and pins. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Inspect with insulation blanket removed.						
52-710-00-01	MRB	51-05-01-210-808 52-05-03-210-814	1.1 1.2	9 YR 18000 FC	8 YR 18000 FC	ALL	ALL
	Inspect forward cargo door skin and structure. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Inspect with insulation blanket removed.						
52-710-00-02	MRB	51-05-01-210-808 52-05-03-210-815	1.1 1.2	9 YR 18000 FC	8 YR 18000 FC	ALL	ALL
	Inspect aft cargo door skin and structure. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Inspect with insulation blanket removed.						
52-730-00-01	MRB	51-05-01-210-809 52-05-03-211-821	1.1 1.2	36 MO 4000 FC	36 MO 4000 FC	ALL	ALL
	Inspect automatic overwing exit door stop fittings and pins. **AIRPLANE NOTE:** Zone 832 and 842 are applicable to 737-800 and 737-900 only. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Inspect with doors opened and lining not removed.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
52-730-00-02	MRB	51-05-01-210-809 52-05-03-211-822	1.1 1.2	36 MO 4000 FC	36 MO 4000 FC	ALL	ALL
	Inspect automatic overwing exit door stop fittings and pins. **AIRPLANE NOTE:** Zone 832 and 842 are applicable to 737-800 and 737-900 only. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Inspect with doors opened and lining not removed.						
52-740-00-01	MRB	51-05-01-210-810 52-05-03-211-823	1.1 1.2	9 YR 18000 FC	8 YR 18000 FC	ALL	ALL
	Inspect the left automatic overwing exit door stop fittings and pins. **AIRPLANE NOTE:** Zone 832 and 842 are applicable to 737-800 and 737-900 only. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Inspect with hatches removed, the door opened or removed. Remove linings and insulations.						
52-740-00-02	MRB	51-05-01-210-810 52-05-03-211-824	1.1 1.2	9 YR 18000 FC	8 YR 18000 FC	ALL	ALL
	Inspect right automatic overwing exit door stop fittings and pins. **AIRPLANE NOTE:** Zone 832 and 842 are applicable to 737-800 and 737-900 only. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Inspect with hatches removed, the door opened or removed. Remove linings and insulations.						
52-760-00-01	MRB	51-05-01-210-808 52-05-03-210-816	1.1 1.2	9 YR 18000 FC	8 YR 18000 FC	ALL	ALL
	Inspect the left automatic overwing exit door skin and structure. **AIRPLANE NOTE:** Zone 832 and 842 are applicable to 737-800 and 737-900 only. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Inspect with hatches removed, door opened or removed. Remove linings and insulations.						
52-760-00-02	MRB	51-05-01-210-808 52-05-03-210-817	1.1 1.2	9 YR 18000 FC	8 YR 18000 FC	ALL	ALL
	Inspect the right automatic overwing exit door skin and structure. **AIRPLANE NOTE:** Zone 832 and 842 are applicable to 737-800 and 737-900 only. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Inspect with hatches removed, the door opened or removed. Remove linings and insulations.						
52-794-00-01	AWL	52-05-02-211-801	1.1	50000 FC	18000 FC	ALL	ALL
	Inspect (Detailed) the lower frame at the stop/guide fitting, forward and aft. See Doc. D626A001-DTR, DTR check form 52-21-04 for alternative inspections. **AIRPLANE NOTE:** Panels 832AZ and 842AZ are applicable to 737-800 and 737-900 only. **ACCESS NOTE:** Removal of the lining and the stop guide is required (note that the stop guide fitting is attached with removable bolts).						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
52-794-00-02	AWL	52-05-02-211-801	1.1	50000 FC	18000 FC	ALL	ALL
	Inspect (Detailed) the lower frame at the stop/guide fitting, forward and aft. See Doc. D626A001-DTR, DTR check form 52-21-04 for alternative inspections. **AIRPLANE NOTE:** Panels 832AZ and 842AZ are applicable to 737-800 and 737-900 only. **ACCESS NOTE:** Removal of the lining and the stop guide is required (note that the stop guide fitting is attached with removable bolts).						
52-794-00-03	AWL	52-05-02-211-801	1.1	50000 FC	18000 FC	ALL	ALL
	Inspect (Detailed) the lower frame at the stop/guide fitting, forward and aft. See Doc. D626A001-DTR, DTR check form 52-21-04 for alternative inspections. **AIRPLANE NOTE:** Panels 832AZ and 842AZ are applicable to 737-800 and 737-900 only. **ACCESS NOTE:** Removal of the lining and the stop guide is required (note that the stop guide fitting is attached with removable bolts).						
52-794-00-04	AWL	52-05-02-211-801	1.1	50000 FC	18000 FC	ALL	ALL
	Inspect (Detailed) the lower frame at the stop/guide fitting, forward and aft. See Doc. D626A001-DTR, DTR check form 52-21-04 for alternative inspections. **AIRPLANE NOTE:** Panels 832AZ and 842AZ are applicable to 737-800 and 737-900 only. **ACCESS NOTE:** Removal of the lining and the stop guide is required (note that the stop guide fitting is attached with removable bolts).						
52-796-00-01	AWL	52-05-02-211-802	1.1	50000 FC	4000 FC	ALL	ALL
	Inspect (Detailed) the area around the fastener locations common to the inner panel and the door frame adjacent to the four (4) door pin locations bounded by LBL 5.70, RBL 14.12, frame STA 325.07 and STA 349.13. See Doc. D626A001-DTR, DTR check form 52-48-04-2 for alternative inspections. **ACCESS NOTE:** Access the interior of the door.						
52-800-00-01	MRB	05-41-08-210-801	1.1	120 DY	120 DY	ALL	ALL
	Perform an external zonal inspection (GV) of the doors. Inspection is accomplished from the ground, without the use of stands or ladders. No additional access panel is required. **AIRPLANE NOTE:** Mid-Exit Doors only on 900ER. **ACCESS NOTE:** Door closed.						
52-802-02-01	MRB	05-41-08-210-802	1.1 1.2	2000 FC 240 DY	2000 FC 240 DY	ALL	ALL
	Perform an external zonal inspection (GV) of the forward cargo door - section 43, sta 460. **INTERVAL NOTE:** Whichever comes first.						
52-804-02-01	MRB	05-41-08-210-803	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the forward cargo door - section 43, sta 460. **INTERVAL NOTE:** Whichever comes first.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
52-806-02-01	MRB	05-41-08-210-804	1.1 1.2	2000 FC 240 DY	2000 FC 240 DY	ALL	ALL
	Perform an external zonal inspection (GV) of the aft cargo door - section 46, sta 827. **INTERVAL NOTE:** Whichever comes first.						
52-808-02-01	MRB	05-41-08-210-805	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the aft cargo door - section 46, sta 827. **INTERVAL NOTE:** Whichever comes first.						
52-810-01-01	MRB	05-41-08-210-806	1.1 1.2	1500 FC 180 DY	1500 FC 180 DY	ALL	ALL
	Perform an external zonal inspection (GV) of the forward passenger door - section 41, sta 345. **INTERVAL NOTE:** Whichever comes first.						
52-812-01-01	MRB	05-41-08-210-807	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the forward passenger door - section 41, sta 345. **INTERVAL NOTE:** Whichever comes first.						
52-814-01-01	MRB	05-41-08-210-808	1.1 1.2	5500 FC 30 MO	5500 FC 30 MO	800 900	ALL
	Perform an external zonal inspection (GV) of the automatic overwing exit - section 44, sta 589.5. **INTERVAL NOTE:** Whichever comes first.						
52-816-01-01	MRB	05-41-08-210-809	1.1 1.2	18000 FC 9 YR	18000 FC 8 YR	800 900	ALL
	Perform an internal zonal inspection (GV) of the automatic overwing exit - section 44, sta 589.5. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Automatic overwing exit door liner removal required.						
52-818-01-01	MRB	05-41-08-210-810	1.1 1.2	5500 FC 30 MO	5500 FC 30 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the automatic overwing exit - section 44, sta 627. **INTERVAL NOTE:** Whichever comes first.						
52-820-01-01	MRB	05-41-08-210-811	1.1 1.2	18000 FC 9 YR	18000 FC 8 YR	ALL	ALL
	Perform an internal zonal inspection (GV) of the automatic overwing exit - section 44, STA 627. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Automatic overwing exit door liner removal required.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
52-822-01-01	MRB	05-41-08-210-812	1.1 1.2	1500 FC 180 DY	1500 FC 180 DY	ALL	ALL
	Perform an external zonal inspection (GV) of the aft passenger door - section 47, sta 980. **INTERVAL NOTE:** Whichever comes first.						
52-824-01-01	MRB	05-41-08-210-813	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the aft passenger door - section 47, sta 980. **INTERVAL NOTE:** Whichever comes first.						
52-826-02-01	MRB	05-41-08-210-816	1.1 1.2	1500 FC 180 DY	1500 FC 180 DY	ALL	ALL
	Perform an external zonal inspection (GV) of the forward galley service door - section 41, sta 340. **INTERVAL NOTE:** Whichever comes first.						
52-828-02-01	MRB	05-41-08-210-817	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the forward galley service door - section 41, STA 340. **INTERVAL NOTE:** Whichever comes first.						
52-830-02-01	MRB	05-41-08-210-818	1.1 1.2	5500 FC 30 MO	5500 FC 30 MO	800 900	ALL
	Perform an external zonal inspection (GV) of the automatic overwing exit - section 44, sta 589.5. **INTERVAL NOTE:** Whichever comes first.						
52-832-02-01	MRB	05-41-08-210-819	1.1 1.2	18000 FC 9 YR	18000 FC 8 YR	800 900	ALL
	Perform an internal zonal inspection (GV) of the automatic overwing exit - section 44, sta 589.5. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Automatic overwing exit door liner removal required.						
52-834-02-01	MRB	05-41-08-210-820	1.1 1.2	5500 FC 30 MO	5500 FC 30 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the automatic overwing exit - section 44, sta 627. **INTERVAL NOTE:** Whichever comes first.						
52-836-02-01	MRB	05-41-08-210-821	1.1 1.2	18000 FC 9 YR	18000 FC 8 YR	ALL	ALL
	Perform an internal zonal inspection (GV) of the automatic overwing exit - section 44, sta 627. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Automatic overwing exit door liner removal required.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
52-838-02-01	MRB	05-41-08-210-822	1.1 1.2	1500 FC 180 DY	1500 FC 180 DY	ALL	ALL
	Perform an external zonal inspection (GV) of the aft galley service door - section 47, sta 980. **INTERVAL NOTE:** Whichever comes first.						
52-840-02-01	MRB	05-41-08-210-823	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the aft galley service door - section 47, sta 980. **INTERVAL NOTE:** Whichever comes first.						
53-010-00-01	MRB	51-05-01-210-801 53-05-03-210-801	1.1 1.2	36 MO 4000 FC	36 MO 4000 FC	ALL	ALL
	Inspect door cutout at forward access door. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Open forward access door.						
53-010-00-02	MRB	51-05-01-210-801 53-05-03-210-802	1.1 1.2	36 MO 4000 FC	36 MO 4000 FC	ALL	ALL
	Inspect door cutout at EE Compartment door. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Open EE compartment door.						
53-020-00-01	MRB	51-05-01-210-806 53-05-03-210-804	1.1	12 YR	6 YR	ALL	ALL
	Inspect nose landing gear wheel well, including canted bulkhead (Sta 224.8 to 227.8), Sta 294.5 bulkhead, side and top panels, trunnion support fitting, actuator support fitting, and drag brace fitting.						
53-030-00-01	MRB	51-05-01-210-809 53-05-03-211-801	1.1 1.2	36 MO 6600 FC	36 MO 6600 FC	ALL	ALL
	Inspect the forward cargo door surround structure, fittings and stops. **INTERVAL NOTE:** Whichever comes first.						
53-030-00-02	MRB	51-05-01-210-809 53-05-03-211-802	1.1 1.2	36 MO 6600 FC	36 MO 6600 FC	ALL	ALL
	Inspect the aft cargo door surround structure, fittings and stops. **INTERVAL NOTE:** Whichever comes first.						
53-050-00-01	MRB	51-05-01-210-806 53-05-03-210-805	1.1	9 YR	3 YR	ALL	ALL
	Inspect main landing gear wheel well, including: 1. Pressure deck web and stiffeners, including attachment to wing center section rear spar at Sta 663; 2. Bulkhead at STA 663; 3. Bulkhead and pressure web at STA 727; 4. Keel beam chords, webs, stiffeners and splice, keel beam/rear spar attachment angles; 5. Stringer 18A web, chord and links; 6. Side strut support frame at STA 706; 7. Main landing gear support frame at STA 695 and 716; 8. Wheel well frame at STA 685; 9. Flap track support fittings.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-060-00-01	MRB	51-05-01-210-806 53-05-03-210-806	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL
	Inspect the forward side of STA 178 bulkhead. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Open nose radome.						
53-070-00-01	MRB	51-05-01-210-808 53-05-03-210-807	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL
	Inspect fuselage lower lobe from STA 178 bulkhead to canted bulkhead (STA 224.8 to 227.8), including bulkheads, skin panels (skins, frames, stringers), longitudinal lap splices, forward access door cutout, and nose wheel well cutout. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Remove weather radar RT mount. Remove/displace insulation blankets as required.						
53-080-00-01	MRB	51-05-01-210-806 53-05-03-210-808	1.1	10 YR	10 YR	ALL	ALL
	Inspect flight compartment floor structure from lower lobe. **ACCESS NOTE:** Access through forward access door						
53-090-00-01	MRB	51-05-01-210-808 53-05-03-210-809	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL
	Inspect fuselage lower lobe from canted bulkhead (Sta 224.8 to 227.8) to Sta 294, including: 1. Skin panels (skins, frames, stringers), longitudinal lap splices, circumferential skin and stringer splices, bulkhead at Sta 259.5; 2. Nose wheel well cutout surround structure, nose wheel well side and top panels; 3. Trunnion support fitting, actuator support fitting and drag brace fitting. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Access through nose wheel well side and top access panels, and through access panel in crew floor. Remove/displace insulation blankets as required.						
53-100-00-01	MRB	51-05-01-210-808 53-05-03-210-810	1.1	12 YR	12 YR	ALL	ALL
	Inspect passenger compartment floor structure in dry areas (away from doors, galleys and lavs) from lower lobe. **ACCESS NOTE:** Remove ceiling and sidewall panels as required. Remove/displace insulation blankets as required. Remove or displace auxiliary fuel tank as required (business jet only). Remove forward airstairs and airstairs compartment (if installed).						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-110-00-01	MRB	51-05-01-210-808 53-05-03-210-811	1.1	NOTE		ALL	ALL
	Inspect passenger compartment floor structure in wet areas (within approximately 20 inches from doors, galleys and lavs) from lower lobe. **INTERVAL NOTE:** Threshold Interval 8YR / Repeat interval 6YR, whichever occurs first, applicable to Airplanes L/N# 1-2412 and those that have not incorporated the HI-TAK Gel Tape. Threshold Interval 9YR / Repeat interval 6YR, whichever occurs first, applicable to Airplanes L/N# 2413 and on, or those that have incorporated the HI-TAK Gel Tape. **ACCESS NOTE:** Remove ceiling and sidewall panels as required. Remove/displace insulation blankets as required. Remove or displace auxiliary fuel tank as required (business jet only). Remove forward airstairs and airstair compartment (if installed. 117BL).						
53-120-00-01	MRB	51-05-01-210-808 53-05-03-210-812	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL
	Inspect EE compartment (STA 294.5 to 396 (STA 400 for 737-700C)), including: 1. Skin panels (skins, frames, stringers), longitudinal lap splices, circumferential skin and stringer splices, bulkhead at STA 294.5; 2. EE compartment door and airstairs door cutout surround structure; 3. Forward entry and galley door cutout surround structure (portion in lower lobe). **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Remove LRUs and racks, do not remove permanently installed structure. Remove/displace insulation blankets as required.						
53-130-00-01	MRB	51-05-01-210-808 53-05-03-210-813	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL
	Inspect forward cargo compartment skin panels including skins, frames, and stringers (note: inspection includes the circumferential skin and stringer splice at STA 500E for the -900 models). **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Remove sidewalls and ceiling panels. Remove/displace insulation blankets as required. Remove/displace auxiliary fuel tank as required (business jet only).						
53-140-00-01	MRB	51-05-01-210-808 53-05-03-210-814	1.1 1.2	8 YR 24000 FC	6 YR 18000 FC	ALL	ALL
	Inspect forward cargo compartment floor structure. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Remove cargo floor panels.						
53-140-00-02	MRB	51-05-01-210-808 53-05-03-210-815	1.1 1.2	8 YR 24000 FC	6 YR 18000 FC	ALL	ALL
	Inspect Aft cargo compartment floor structure. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Remove cargo floor panels.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-150-00-01	MRB	51-05-01-210-808 53-05-03-211-803	1.1 1.2	9 YR 24000 FC	6 YR 18000 FC	ALL	ALL
	Inspect forward cargo door cutout surround structure. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Remove door reveals. Remove sidewalls as required. Remove/displace insulation blankets as required.						
53-160-00-01	MRB	51-05-01-210-808 53-05-03-210-816	1.1 1.2	8 YR 24000 FC	6 YR 18000 FC	ALL	ALL
	Inspect forward bilge skin panels including skins, frames, stringers, longitudinal lap splices, and cargo door cutout surround structure in bilge (note: inspection includes the circumferential skin and stringer splice at STA 500E for the -900 models). **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Remove cargo floor panels and scuff plates. Remove/Displace insulation blankets as required.						
53-170-00-01	MRB	51-05-01-210-808 53-05-03-210-817	1.1 1.2	12 YR 36000 FC	10 YR 36000 FC	ALL	ALL
	Inspect area aft of forward cargo compartment, including skin panels (skins, frames, stringers), longitudinal lap splices, circumferential skin and stringer splices, and forward side of Sta 540 bulkhead and bulkhead splices. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Remove forward cargo compartment aft bulkhead panels. Remove/displace insulation blankets as required. Remove ducting as required.						
53-180-00-01	MRB	51-05-01-210-806 53-05-03-210-818	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL
	Inspect area under lower wing-to-body fairing (forward of wing box), including skin panels, longitudinal lap splices, keel beam extension, wing-to-body drag angles, and Sta 540 bulkhead. **INTERVAL NOTE:** Whichever comes first.						
53-190-00-01	MRB	51-05-01-210-808 53-05-03-210-819	1.1 1.2	12 YR 36000 FC	10 YR 36000 FC	ALL	ALL
	Inspect fuselage lower lobe above wing box center section upper panel, including side skin panels (skins, frames and stringers), Sta 540 bulkhead, and overwing frames and stub beams. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Remove floor panels. Remove/displace insulation blankets as required.						
53-200-00-01	MRB	51-05-01-210-808 53-05-03-210-820	1.1 1.2	12 YR 36000 FC	10 YR 36000 FC	ALL	ALL
	Inspect fuselage lower lobe above main landing gear wheel well, including: 1. Pressure deck web to stiffeners, stiffener attachment to floor beam at STA 727; 2. Side skin panels, circumferential skin and stringer splice; 3. Bulkheads at STA 663 and 727; 4. Side strut support frame at STA 706; 5. Main landing gear support frame at STA 695 and 716; 6. Wheel well frame at STA 685. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Remove floor panels. Remove/displace insulation blankets as required.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-210-00-01	MRB	51-05-01-210-804 53-05-03-210-821	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL

Inspect keel beam under wing-to-body fairing (under wing box, Sta 540 to 663.75), including keel beam chords, webs, stiffeners, splice, keel beam/rear spar attachment angles.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Option 1: Remove center wing-to-body fairing (192CL, 192CR, 192E, 192F) and open AC bay access door. Option 2: Open AC bay door and remove AC pack to gain access to access holes.

53-220-00-01	MRB	51-05-01-210-806 53-05-03-210-823	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL

Inspect keel beam in wheel well (Sta 663.75 to 727), including keel beam chords, webs, stiffeners, splice, keel beam/rear spar attachment angles.

INTERVAL NOTE: Whichever comes first.

53-230-00-01	MRB	51-05-01-210-808 53-05-03-210-824	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL

Inspect aft cargo compartment, including: 1. Side skin panels (skin, frames, stringers), circumferential skin and stringer splices, (note: located at Sta 727I for -900 and 727L for -900ER models); 2. Stringer 18 strap at side of body; 3. Stringer 18A web, chord and links; 4. Aft side of STA 727 bulkhead and pressure web.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Remove sidewall and ceiling panels, E6 LRU, access panels around vacuum lav tank. Remove/displace insulation blankets as required. Remove/displace vacuum lav components as required. Remove/displace auxiliary fuel tank as required (business jet only).

53-240-00-01	MRB	51-05-01-210-808 53-05-03-211-804	1.1 1.2	8 YR 24000 FC	6 YR 18000 FC	ALL	ALL

Inspect aft cargo door cutout surround structure.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Remove door reveals. Remove sidewalls as required. Remove/displace insulation blankets as required.

53-250-00-01	MRB	51-05-01-210-808 53-05-03-210-825	1.1 1.2	8 YR 24000 FC	6 YR 18000 FC	ALL	ALL

Inspect aft bilge skin panels (skins, frames, stringers), longitudinal lap splices, circumferential skin and stringer splices, (note: located at Sta 727I for -900 and 727L for -900ER models); Sta 727 bulkhead and pressure web, and cargo door cutout surround structure in bilge.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Remove cargo floor panels and scuff plates. Remove/Displace insulation blankets as required.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-260-00-01	MRB	51-05-01-210-808 53-05-03-210-826	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL

Inspect area aft of cargo compartment, including: 1. Skin panels (skins, frames, stringers), longitudinal lap splices, circumferential skin and stringer splices; 2. Aft entry and galley door cutout surround structure in lower lobe; 3. STA 1016 bulkhead, including chords, pressure web, stiffeners, chord/web attachments; 4. Stringer splice fittings and tension bolts at STA 1016.

AIRPLANE NOTE: Task not applicable to -900ER and -800 with Flat Pressure Bulkhead installed.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Remove aft cargo compartment aft bulkhead panel and potable water tank.
Remove/displace insulation blankets as required.

53-270-00-01	MRB	51-05-01-210-808 53-05-03-210-827	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL

Inspect area under lower wing-to-body fairing (aft of wheel well), including skin panels, longitudinal lap splices, circumferential skin splice, stringer 18 strap at side of body, stringer 18A (web, chords and links), and keel beam extension.

INTERVAL NOTE: Whichever comes first.

53-280-00-01	MRB	51-05-01-210-806 53-05-03-210-828	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL

Inspect area under above-wing wing-to-body fairing, including skin panels, circumferential skin splices, and stringer 18 strap at side of body.

INTERVAL NOTE: Whichever comes first.

53-290-00-01	MRB	51-05-01-210-809 53-05-03-210-829	1.1 1.2	9 YR 24000 FC	8 YR 24000 FC	ALL	ALL

Inspect the automatic overwing exit cutout structure, fittings and stops;

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Open automatic overwing exits.

53-310-00-01	MRB	51-05-01-210-809 53-05-03-211-805	1.1 1.2	36 MO 6600 FC	36 MO 6600 FC	ALL	ALL

Inspect door frames, stops, latches and hinges on the forward door cutout surround structure.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Open forward entry door.

53-310-00-02	MRB	51-05-01-210-809 53-05-03-211-806	1.1 1.2	36 MO 6600 FC	36 MO 6600 FC	ALL	ALL

Inspect door frames, stops, latches and hinges on forward galley door cutout surround structure.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Open forward galley service door.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-310-00-03	MRB	51-05-01-210-809 53-05-03-211-807	1.1 1.2	36 MO 6600 FC	36 MO 6600 FC	ALL	ALL

Inspect door frames, stops, latches and hinges on aft entry cutout surround structure.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Open aft entry door.

53-310-00-04	MRB	51-05-01-210-809 53-05-03-211-808	1.1 1.2	36 MO 6600 FC	36 MO 6600 FC	ALL	ALL

Inspect door frames, stops, latches and hinges on aft galley door cutout surround structure.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Open aft galley door.

53-330-00-01	MRB	51-05-01-210-808 53-05-03-210-830	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL

Inspect flight compartment, including skin panels (skins, frames, stringers), circumferential skin and stringer splice, crew cabin window cutout structure, and structure adjacent to ground block behind rudder pedal. Inspection area does not include: Forward side of frame at STA 259.5 and structure 3 inches forward of STA 259.5; BL 0 + 4 inches (Left and Right); Forward and aft side of Frame 259.5 and structure from STA 249 to STA 263 between floor and S-5L (excluding window and window frame structure); Forward side of Frame 259.5 and structure 3 inches forward of STA 259.5 between floor and S-3R; Structure forward of STA 203.8 to STA 178 and from floor up to window frame (except structure adjacent to ground block behind rudder pedal); skin, frames and stringers above P5 panel.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Remove glare shield, liners, overhead units and panels as required. Remove/displace insulation blankets as required.

53-335-00-01	MRB	51-05-01-210-808 53-05-03-210-861	1.1 1.2	20 YR 50000 FC	8 YR 24000 FC	ALL	ALL

Inspect flight compartment, including skin panels (skins, frames, stringers), circumferential skin and stringer splice, crew cabin window cutout structure, and forward pressure bulkhead within the following areas: Forward side of frame at STA 259.5 and structure 3 inches forward of STA 259.5; BL 0 + 4 inches (Left and Right); Forward and aft side of Frame 259.5 and structure from STA 249 to STA 263 between floor and S-5L (excluding window and window frame structure); Forward side of Frame 259.5 and structure 3 inches forward of STA 259.5 between floor and S-3R; Structure forward of STA 203.8 to STA 178 and from floor up to window frame; and skin, frames and stringers above P5 panel.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Remove glare shield, liners, overhead units and panels as required. Remove/displace insulation blankets as required.

53-340-00-01	MRB	51-05-01-210-808 53-05-03-210-831	1.1	10 YR	10 YR	ALL	ALL

Inspect flight compartment floor structure.

ACCESS NOTE: Remove sidewalls and floor panels as required. Remove/displace insulation blankets as required.

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-350-00-01	MRB	51-05-01-210-808 53-05-03-210-832	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL

Inspect passenger compartment from Sta 270 to 360 (except areas around door cutouts), including skin panels (skins, frames, stringers), longitudinal lap splices, circumferential skin and stringer splices.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Remove galleys/lavs. Remove cabin interior as required. Remove/displace insulation blankets as required.

53-360-00-01	MRB	51-05-01-210-808 53-05-03-211-809	1.1	NOTE		ALL	ALL

Inspect forward entry door cutout surround structure (the door cutout to, and including, the door side of the first frame from the door in both the forward and aft directions).

INTERVAL NOTE: Threshold interval 8YR / 24000 FC and Repeat interval 6YR / 18000 FC (Whichever comes first), applicable to Airplanes L/N# 1-5645 and L/N# 5653-5659 and those that have not incorporated the doorsill corrosion protection/enhanced moisture barrier. Threshold interval 9YR / 24000 FC and Repeat interval 6YR / 18000 FC (Whichever comes first), applicable to Airplanes L/N# 5646-5652 and L/N# 5660 and on, or those that have incorporated the doorsill corrosion protection/enhanced moisture barrier.

ACCESS NOTE: Remove galleys/lavs. Remove cabin interior as required. Remove/displace insulation blankets as required.

53-360-00-02	MRB	51-05-01-210-808 53-05-03-211-810	1.1	NOTE		ALL	ALL

Inspect forward galley door cutout surround structure (the door cutout to, and including, the door side of the first frame from the door in both the forward and aft directions).

INTERVAL NOTE: Threshold interval 8YR / 24000 FC and Repeat interval 6YR / 18000 FC (Whichever comes first), applicable to Airplanes L/N# 1-5645 and L/N# 5653-5659 and those that have not incorporated the doorsill corrosion protection/enhanced moisture barrier. Threshold interval 9YR / 24000 FC and Repeat interval 6YR / 18000 FC (Whichever comes first), applicable to Airplanes L/N# 5646-5652 and L/N# 5660 and on, or those that have incorporated the doorsill corrosion protection/enhanced moisture barrier.

ACCESS NOTE: Remove galleys/lavs. Remove cabin interior as required. Remove/displace insulation blankets as required.

53-360-00-03	MRB	51-05-01-210-808 53-05-03-211-811	1.1	NOTE		ALL	ALL

Inspect aft entry door cutout surround structure (the door cutout to, and including, the door side of the first frame from the door in both the forward and aft directions).

INTERVAL NOTE: Threshold interval 8YR / 24000 FC and Repeat interval 6YR / 18000 FC (Whichever comes first), applicable to Airplanes L/N# 1-5645 and L/N# 5653-5659 and those that have not incorporated the doorsill corrosion protection/enhanced moisture barrier. Threshold interval 9YR / 24000 FC and Repeat interval 6YR / 18000 FC (Whichever comes first), applicable to Airplanes L/N# 5646-5652 and L/N# 5660 and on, or those that have incorporated the doorsill corrosion protection/enhanced moisture barrier.

ACCESS NOTE: Remove galleys/lavs. Remove cabin interior as required. Remove/displace insulation blankets as required.

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-360-00-04	MRB	51-05-01-210-808 53-05-03-211-812	1.1	NOTE		ALL	ALL

Inspect aft galley door cutout surround structure (the door cutout to, and including, the door side of the first frame from the door in both the forward and aft directions).

INTERVAL NOTE: Threshold interval 8YR / 24000 FC and Repeat interval 6YR / 18000 FC (Whichever comes first), applicable to Airplanes L/N# 1-5645 and L/N# 5653-5659 and those that have not incorporated the doorsill corrosion protection/enhanced moisture barrier. Threshold interval 9YR / 24000 FC and Repeat interval 6YR / 18000 FC (Whichever comes first), applicable to Airplanes L/N# 5646-5652 and L/N# 5660 and on, or those that have incorporated the doorsill corrosion protection/enhanced moisture barrier.

ACCESS NOTE: Remove galleys/lavs. Remove cabin interior as required. Remove/displace insulation blankets as required.

53-370-00-01	MRB	51-05-01-210-808 53-05-03-210-833	1.1	12 YR	12 YR	ALL	ALL

Inspect passenger compartment floor structure in dry area (away from doors, galleys and lavs). Exclude floor structure from Sta 540 to 727.

NOTE: Not applicable to airplanes with flat aft pressure bulkhead.

ACCESS NOTE: Remove floor panels and sidewalls as required. Remove/displace insulation blankets as required.

53-380-00-01	MRB	51-05-01-210-808 53-05-03-210-834	1.1	NOTE		ALL	ALL

Inspect passenger compartment floor structure in wet area (within approximately 20 inches from doors, galleys and lavs, and the floor structure below the door to, and including, the door side of the first frame from the door in both the forward and aft directions).

NOTE: Not applicable to airplanes with flat aft pressure bulkhead.

INTERVAL NOTE: Threshold interval 8YR / Repeat interval 6YR, whichever comes first, applicable to Airplanes L/N# 1-2412 and those that have not incorporated the HI-TAK Gel Tape. Threshold interval 9YR / Repeat interval 6YR, whichever comes first, applicable to Airplanes L/N# 2413 and on, or those that have incorporated the HI-TAK Gel Tape.

ACCESS NOTE: Remove galleys and lavs. Remove floor panels and sidewalls as required. Remove/displace insulation blankets as required.

53-390-00-01	MRB	51-05-01-210-804 53-05-03-210-835	1.1	9 YR	6 YR	ALL	ALL

Inspect galley and lav attach fittings and any other easily visible portions of the floor structure in wet area (within approximately 20 inches from galleys and lavs, and the floor structure below the door to, and including, the door side of the first frame from the door in both the forward and aft directions).

ACCESS NOTE: Galleys and lavs removal is not required. Remove galley kick-plates and any other easily removable panels that may help inspect areas under galleys and lavs.

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-400-00-01	MRB	51-05-01-210-808 53-05-03-210-836	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL

Inspect passenger compartment from STA 360 to 663.75, including: 1. Skin panels (skins, frames and stringers), longitudinal lap splices, circumferential skin and stringer splices; 2. Window belt structure; 3. Overwing emergency exit cutout structure; 4. Forward cargo door cutout surround structure (portion in upper lobe); 5. STA 540 and 663 bulkheads and splices; 6. Overwing frames and stub beams.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Remove cabin interior as required. Remove/displace insulation blankets as required.

53-410-00-01	MRB	51-05-01-210-808 53-05-03-210-837	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL

Inspect passenger compartment from STA 663.75 to 1016 (except areas around door cutouts), including: 1. Skin panels (skins, frames and stringers), longitudinal lap splices, circumferential skin and stringer splices (note: inspection includes the circumferential skin and stringer splice at Sta 727I for the -900 models); 2. Window belt structure; 3. STA 663 bulkhead and splices; 4. STA 727 bulkhead; 5. Side strut support frame at STA 706; 6. Main landing gear support frames at STA 695 and 716; 7. Wheel well frame at STA 685; 8. Aft cargo door cutout surround structure (portion in upper lobe); 9. Forward side of STA 1016 bulkhead (chords, pressure web, stiffeners, chord/web attachments), including vertical fin front spar fittings; 10. Stringer splice fittings and tension bolts at STA 1016.

AIRPLANE NOTE: Task not applicable to -900ER and -800 with Flat Pressure Bulkhead installed.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Remove galleys/lavs. Remove cabin interior as required. Remove/replace insulation blankets as required.

53-420-00-01	MRB	51-05-01-210-806 53-05-03-210-838	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL

Inspect area aft of STA 1016 pressure bulkhead to STA 1088, including: 1. Skin panels (skins, frames and stringers), longitudinal lap splices, circumferential skin and stringer splices; 2. Aft side of STA 1016 bulkhead (chords, pressure web, stiffeners, chord/web attachments); 3. Stringer splice fittings and tension bolts at STA 1016; 4. STA 1088 bulkhead, including vertical fin rear spar fittings and horizontal stabilizer center section jackscrew fitting lugs and bolts.

AIRPLANE NOTE: Task not applicable to -900ER and -800 with Flat Pressure Bulkhead installed.

INTERVAL NOTE: Whichever comes first.

53-430-00-01	MRB	51-05-01-210-804 53-05-03-210-839	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL

Inspect stabilizer torsion box compartment and APU compartment, including: 1. Skin panels (skins, frames and stringers), longitudinal lap splices; 2. STA 1088 bulkhead, including vertical fin rear spar fittings; 3. Forward side of STA 1156 bulkhead, including horizontal stabilizer hinge fittings and bolts; 4. Upper horizontal deck (at stringer 6) and lower horizontal deck (at stringer 12).

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: For area below stringer 12, remove APU and firewalls; remove APU plenum as required. For area above stringer 12, adjust stabilizer trim as required. For access to Sta 1156 horizontal stabilizer hinge fitting lugs and bolts, remove gap seal and horizontal stabilizer rear spar sliding seal as required.

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-440-00-01	MRB	51-05-01-210-806 53-05-03-210-840	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL
	Inspect aft side of STA 1156 bulkhead. **INTERVAL NOTE:** Whichever comes first.						
53-450-00-01	MRB	51-05-01-210-806 53-05-03-210-841	1.1 1.2	9 YR 24000 FC	8 YR 24000 FC	ALL	ALL
	Inspect fuselage skin under dorsal fin and aft to Sta 1016, including circumferential splice. **INTERVAL NOTE:** Whichever comes first.						
53-460-00-01	MRB	51-05-01-210-806 53-05-03-210-842	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL
	Inspect vertical fin front spar fitting lugs and bolts (Sta 1016). **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Pin removal is not required.						
53-470-00-01	MRB	51-05-01-210-804 53-05-03-210-843	1.1	20 YR	8 YR	ALL	ALL
	Inspect vertical fin front spar fitting lugs and bolts (STA 1016). **ACCESS NOTE:** Pin removal is required. Remove only one pin at a time.						
53-480-00-01	MRB	51-05-01-210-806 53-05-03-210-844	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL
	Inspect vertical fin rear spar fitting lugs and bolts at Sta 1088. Inspect fuselage skin under vertical fin from Sta 1016 to 1088. **INTERVAL NOTE:** Whichever comes first.						
53-510-00-01	MRB	51-05-01-210-808 53-05-03-210-862	1.1 1.2	9 YR 24000 FC	9 YR 24000 FC	800 900ER	ALL
	Inspect the skin and lugs under the antenna base plate Sta. 727D to 727H+5, S-4L to S-4R. **AIRPLANE NOTE:** Applicable to airplanes with a KU antenna radome installed at Sta. 727D to 727H+5, S-4L to S-4R (737-800 and 737-900ER Only). **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Antenna radome and adapter plate removal required.						
53-600-00-01	AWL	53-05-02-250-801	1.1	50000 FC	36000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (High Frequency Eddy Current) the upper skin along the upper fastener row at stringers S-4L and S-4R from STA 259.5 to STA 360. See Doc. D626A001-DTR, DTR check form 53-10-03-1, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-41.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-600-00-02	AWL	53-05-02-250-801	1.1	50000 FC	36000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (High Frequency Eddy Current) the upper skin along the upper fastener row at stringers S-4L and S-4R from STA 259.5 to STA 360. See Doc. D626A001-DTR, DTR check form 53-10-03-1, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-41.						
53-600-20-01	AWL	53-05-02-250-803	1.1	50000 FC	36000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (Low Frequency Eddy Current) the lower skin along the lower fastener row at stringers S-4L and S-4R from STA 259.5 to STA 360. See Doc. D626A001-DTR, DTR check form 53-10-03-2, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-50.						
53-600-20-02	AWL	53-05-02-250-803	1.1	50000 FC	36000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (Low Frequency Eddy Current) the lower skin along the lower fastener row at stringers S-4L and S-4R from STA 259.5 to STA 360. See Doc. D626A001-DTR, DTR check form 53-10-03-2, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-50.						
53-600-30-01	AWL	53-05-02-250-805	1.1	50000 FC	36000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the upper skin along the upper fastener row at stringer S-14L (from STA 259.5 to STA 294.5, and from STA 350 to STA 360) and at stringer S-14R (from STA 259.5 to STA 277, and from STA 344 to STA 360). See Doc D626A001-DTR, DTR check form 53-10-03-3 for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-41.						
53-600-30-02	AWL	53-05-02-250-805	1.1	50000 FC	36000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the upper skin along the upper fastener row at stringer S-14L (from STA 259.5 to STA 294.5, and from STA 350 to STA 360) and at stringer S-14R (from STA 259.5 to STA 277, and from STA 344 to STA 360). See Doc D626A001-DTR, DTR check form 53-10-03-3 for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-41.						
53-600-40-01	AWL	53-05-02-250-806	1.1	50000 FC	36000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) the lower skin along the lower fastener row at stringer S-14L (from STA 259.5 to STA 294.5 and from STA 350 to STA 360) and at stringer S-14R (from STA 259.5 to STA 277 and from STA 344 to STA 360). See Doc D626A001-DTR, DTR check form 53-10-03-4 for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-50.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-600-40-02	AWL	53-05-02-250-806	1.1	50000 FC	36000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) the lower skin along the lower fastener row at stringer S-14L (from STA 259.5 to STA 294.5 and from STA 350 to STA 360) and at stringer S-14R (from STA 259.5 to STA 277 and from STA 344 to STA 360). See Doc D626A001-DTR, DTR check form 53-10-03-4 for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-50.						
53-600-50-01	AWL	53-05-02-250-807	1.1	50000 FC	36000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) the upper skin along the upper fastener row at stringer S-24L (from STA 259.5 to STA 334) and at stringer S-24R (from STA 259.5 to STA 360). See Doc D626A001-DTR, DTR check form 53-10-03-5 for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-50.						
53-600-50-02	AWL	53-05-02-250-807	1.1	50000 FC	36000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) the upper skin along the upper fastener row at stringer S-24L (from STA 259.5 to STA 334) and at stringer S-24R (from STA 259.5 to STA 360). See Doc D626A001-DTR, DTR check form 53-10-03-5 for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-50.						
53-601-01-01	AWL	53-05-02-250-809	1.1	50000 FC	12000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the AB post structure, from inside the aircraft, between the upper to lower sills. See Doc D626A001-DTR, DTR check form 53-10-04-1a for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-75. **AIRPLANE NOTE:** Applicable to airplanes L/N 1389 and on.						
53-601-21-01	AWL	53-05-02-211-802	1.1	50000 FC	4000 FC	ALL	ALL
	Inspect (Detailed) the CD post, from outside the aircraft, along the entire post length on both the left and right sides. See Doc D626A001-DTR, DTR check form 53-10-04-2a for alternative inspections. **AIRPLANE NOTE:** Applicable to airplanes L/N 1389 and on.						
53-601-21-02	AWL	53-05-02-211-802	1.1	50000 FC	4000 FC	ALL	ALL
	Inspect (Detailed) the CD post, from outside the aircraft, along the entire post length on both the left and right sides. See Doc D626A001-DTR, DTR check form 53-10-04-2a for alternative inspections. **AIRPLANE NOTE:** Applicable to airplanes L/N 1389 and on.						
53-601-30-01	AWL	53-05-02-211-803	1.1	50000 FC	18000 FC	ALL	ALL
	Inspect (Detailed) the EF post, from outside the aircraft, along the entire post length on both the left and right sides. See Doc D626A001-DTR, DTR check form 53-10-04-3 for alternative inspections.						
53-601-30-02	AWL	53-05-02-211-803	1.1	50000 FC	18000 FC	ALL	ALL
	Inspect (Detailed) the EF post, from outside the aircraft, along the entire post length on both the left and right sides. See Doc D626A001-DTR, DTR check form 53-10-04-3 for alternative inspections.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-601-41-01	AWL	53-05-02-250-811	1.1	50000 FC	4000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) both rows of fasteners attaching the skin to the BD Sill, from outside the aircraft, between LBL 8 and LBL 26.5. Repeat the process between RBL 8 and RBL 26.5.

See Doc D626A001-DTR, DTR check form 53-10-04-4a for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-76.

AIRPLANE NOTE: Applicable to airplanes L/N 1389 and on.

53-601-41-02	AWL	53-05-02-250-811	1.1	50000 FC	4000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) both rows of fasteners attaching the skin to the BD Sill, from outside the aircraft, between LBL 8 and LBL 26.5. Repeat the process between RBL 8 and RBL 26.5.

See Doc D626A001-DTR, DTR check form 53-10-04-4a for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-76.

AIRPLANE NOTE: Applicable to airplanes L/N 1389 and on.

53-601-50-01	AWL	53-05-02-250-812	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the flanges of the Point "D" Fitting, from the CD post inboard to the second fastener common to the BD sill, on both the left and right sides.

See Doc D626A001-DTR, DTR check form 53-10-04-5 for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-72.

ACCESS NOTE: Remove glareshield as required to perform inspection.

53-601-50-02	AWL	53-05-02-250-812	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the flanges of the Point "D" Fitting, from the CD post inboard to the second fastener common to the BD sill, on both the left and right sides.

See Doc D626A001-DTR, DTR check form 53-10-04-5 for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-72.

ACCESS NOTE: Remove glareshield as required to perform inspection.

53-601-60-01	AWL	53-05-02-250-813	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the Inconel angle around the seven fasteners that join the angle to the B-D Sill Web and Point "D" Fitting.

See Doc D626A001-DTR, DTR check form 53-10-04-6 for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-71.

ACCESS NOTE: Remove glareshield if/as required to perform inspection. There are three (3) fasteners that join the angle, web and fitting. There are four (4) fasteners that join the angle and the fitting.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-601-60-02	AWL	53-05-02-250-813	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the Inconel angle around the seven fasteners that join the angle to the B-D Sill Web and Point "D" Fitting.
See Doc D626A001-DTR, DTR check form 53-10-04-6 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-71.

ACCESS NOTE: Remove glareshield if/as required to perform inspection. There are three (3) fasteners that join the angle, web and fitting. There are four (4) fasteners that join the angle and the fitting.

53-602-10-01	AWL	53-05-02-250-814	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the subsurface of the first row of fasteners, on the left and right side, of BL 0.0 between the cab window cutout and STA 259.5 panel splice.
See Doc D626A001-DTR, DTR check form 53-10-05-1 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-57.

53-602-10-02	AWL	53-05-02-250-814	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the subsurface of the first row of fasteners, on the left and right side, of BL 0.0 between the cab window cutout and STA 259.5 panel splice.
See Doc D626A001-DTR, DTR check form 53-10-05-1 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-57.

53-602-20-01	AWL	53-05-02-211-862	1.1	50000 FC	4000 FC	ALL	ALL

Inspect (Detailed) the skin around all of the fastener locations from stringer S-10L to S-10R, from STA 259.5 to STA 360, except at the lap splices and antennas. (53-10-08-1).
See Doc D626A001-DTR, DTR check form 53-30-01-2 for alternative inspections.

53-605-10-01	AWL	53-05-02-211-804	1.1	50000 FC	4000 FC	ALL	ALL

Inspect (Detailed) the skin at the edge of the E/E door cutout and around the two rows of fasteners adjacent to the edge. Inspection is performed along the entire perimeter of the E/E door cutout and bounded by STA 323.7 and STA 351.2, and RBL 15.47 and LBL 6.74.
See Doc D626A001-DTR, DTR check form 53-10-13-1 for alternative inspections.

53-606-40-01	AWL	53-05-02-250-816	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the two rows of fasteners common to the forward edge frame and skin at STA 303.9 from stringers S-11L and S-12L.
See Doc. D626A001-DTR, DTR check form 53-10-14-4 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-89.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-606-50-01	AWL	53-05-02-250-817	1.1	50000 FC	36000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) around the two rows of fasteners common to the forward edge frame and skin at STA 303.9 from stringers S-7L to S-11L and stringers S-12L to S-13L. See Doc. D626A001-DTR, DTR check form 53-10-14-5 for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-89.						
53-606-70-01	AWL	53-05-02-211-805	1.1	50000 FC	9000 FC	ALL	ALL
	Inspect (Detailed) around the fasteners in the inboard flange of the stop fittings from stringers S-7 through S-16. See Doc. D626A001-DTR, DTR check form 53-10-14-7 for alternative inspections. **ACCESS NOTE:** Remove interior panels as required to perform inspection.						
53-618-00-01	AWL	53-05-02-250-818	1.1	50000 FC	30000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) around the four fasteners on each stop strap at stringer S-7 thru S-14. See Doc. D626A001-DTR, DTR check form 53-10-14-8 for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-11-01. **ACCESS NOTE:** Remove interior panels as required to perform inspection.						
53-619-00-01	AWL	53-05-02-250-819	1.1	50000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) around the first two (2) fasteners in the necked down section of the stop straps at S-15 and S-16. See Doc. D626A001-DTR, DTR check form 53-10-14-9 for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-11-01. **ACCESS NOTE:** Remove interior panels as required to perform inspection.						
53-620-00-01	AWL	53-05-02-211-807	1.1	50000 FC	24000 FC	600 700 800 900 900ER	ALL
	Inspect (Detailed) the S-15 and S-16 tension straps at the fastener holes on either side of STA 351.2 frame. See Doc. D626A001-DTR, DTR check form 53-10-14-10 for alternative inspections. **ACCESS NOTE:** Open FWD Entry Door. Removal of interior panel is required to perform the inspection.						
53-621-00-01	AWL	53-05-02-211-808	1.1	50000 FC	18000 FC	ALL	ALL
	Inspect (Detailed) the Aft frame stops at the inner flange holes near STA 348.2 from stringer S-7 to S-14. See Doc. D626A001-DTR, DTR check form 53-10-14-11 for alternative inspections. **ACCESS NOTE:** Open FWD Entry Door. Removal of interior panel is required to perform the inspection.						
53-622-00-01	AWL	53-05-02-250-820	1.1	50000 FC	36000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the skin around the fastener holes and along the edge of the cutout hidden by the scuff plates from STA 303 to STA 350. See Doc. D626A001-DTR, DTR check form 53-10-14-12 for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-43. **ACCESS NOTE:** Removal of scuff plate is required to perform the inspection.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-623-00-01	AWL	53-05-02-250-931	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the skin around the fastener holes and along the edge of the cutout hidden by the scuff plates from STA 303 to STA 350. (53-10-15).
See Doc. D626A001-DTR, DTR check form 53-10-14-12 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-43.

ACCESS NOTE: Removal of scuff plate is required to perform this inspection.

53-624-00-01	AWL	53-05-02-250-821	1.1	50000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the four (4) fastener at each stop location common to the intercostal tension strap at the forward edge frame at stops #1, #2, #5, #6.
See Doc. D626A001-DTR, DTR check form 53-10-15-2 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-98.

ACCESS NOTE: Remove interior panels as required to perform inspection. Door stops are numbered from the bottom up.

53-625-00-01	AWL	53-05-02-250-822	1.1	50000 FC	9000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) around fasteners common to the skin and chords between stringers S-8R and S-14R at the forward and aft edge frames at STA 291.5 and STA 328.5.
See Doc. D626A001-DTR, DTR check form 53-10-15-4 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-55.

53-626-00-01	AWL	53-05-02-250-823	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) around the fasteners in the WL 172 beam just outboard of LBL 17 and RBL 17 at the Nose Wheel Well AFT bulkhead.
See Doc. D626A001-DTR, DTR check form 53-10-18-1 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-99.

ACCESS NOTE: Access through E/E Bay Aft of Nose Wheel Well.

53-626-00-02	AWL	53-05-02-250-823	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) around the fasteners in the WL 172 beam just outboard of LBL 17 and RBL 17 at the Nose Wheel Well AFT bulkhead.
See Doc. D626A001-DTR, DTR check form 53-10-18-1 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-99.

ACCESS NOTE: Access through E/E Bay Aft of Nose Wheel Well.

53-627-00-01	AWL	53-05-02-211-809	1.1	50000 FC	24000 FC	600 700 800 900 900ER	ALL

Inspect (Detailed) the AFT access cutout forward vertical beam at STA 260, from WL 170 to WL 184.
See Doc. D626A001-DTR, DTR check form 53-10-19-4 for alternative inspections.

ACCESS NOTE: Access through Left Aft Nose Wheel Well Panel.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-627-00-02	AWL	53-05-02-211-809	1.1	50000 FC	24000 FC	600 700 800 900 900ER	ALL

Inspect (Detailed) the AFT access cutout forward vertical beam at STA 260, from WL 170 to WL 184.
See Doc. D626A001-DTR, DTR check form 53-10-19-4 for alternative inspections.

ACCESS NOTE: Access through Right Aft Nose Wheel Well Panel.

53-628-00-01	AWL	53-05-02-211-810	1.1	50000 FC	17000 FC	ALL	ALL

Inspect (Detailed) the inboard and outboard drag brace fittings around the perimeter of the bushings at STA 262, BL 16, WL 189.3.
See Doc. D626A001-DTR, DTR check form 53-10-20-2 for alternative inspections.

ACCESS NOTE: For Direction 1, removal of Drag Brace is required.

53-628-00-02	AWL	53-05-02-211-810	1.1	50000 FC	17000 FC	ALL	ALL

Inspect (Detailed) the inboard and outboard drag brace fittings around the perimeter of the bushings at STA 262, BL 16, WL 189.3.
See Doc. D626A001-DTR, DTR check form 53-10-20-2 for alternative inspections.

ACCESS NOTE: For Direction 1, removal of Drag Brace is required.

53-629-00-01	AWL	53-05-02-211-811	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (Detailed) the inboard and outboard fitting segments of the Trunnion Support Fitting around the pin socket at BS 294.5, WL 156.1, and BL 16.
See Doc. D626A001-DTR, DTR check form 53-10-20-3 for alternative inspections.

53-629-00-02	AWL	53-05-02-211-811	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (Detailed) the inboard and outboard fitting segments of the Trunnion Support Fitting around the pin socket at BS 294.5, WL 156.1, and BL 16.
See Doc. D626A001-DTR, DTR check form 53-10-20-3 for alternative inspections.

53-630-00-01	AWL	53-05-02-211-812	1.1	50000 FC	9000 FC	ALL	ALL

Inspect (Detailed) the upper and lower outer sill chords from STA 303.8 to STA 348.
See Doc. D626A001-DTR, DTR check form 53-10-21-1 for alternative inspections.

AIRPLANE NOTE: Applicable to airplanes with airstairs installed.

ACCESS NOTE: Opening of Air Stair Door is required to perform this inspection.

53-631-00-01	AWL	53-05-02-250-824	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the forward and aft edge frame webs around the fasteners common to the corner clips at STA 303.9 and STA 351.2.
See Doc. D626A001-DTR, DTR check form 53-10-21-4 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-11-27.

AIRPLANE NOTE: Applicable to airplanes with airstairs installed.

ACCESS NOTE: Opening of Air Stair Door is required to perform this inspection.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-633-00-01	AWL	53-05-02-211-813	1.1	50000 FC	4000 FC	ALL	ALL
	Inspect (Detailed) the skin around all of the fastener locations from stringers S-10L to S-10R, from STA 360 to STA 540, except at the lap splices and antennas. See Doc D626A001-DTR, DTR check form 53-30-01-2 for alternative inspections.						
53-634-00-01	AWL	53-05-02-211-814	1.1	50000 FC	36000 FC	ALL	ALL
	Inspect (Detailed) the exterior surface of the skin under the TCAS Antenna at STA 385, the ATC Antenna at STA 430, and the GPS Antenna at STA 500A. See Doc D626A001-DTR, DTR check form 53-30-01-4 for alternative inspections. **AIRPLANE NOTE:** For the 737-600, GPS Antennas are located at STA 482A. **ACCESS NOTE:** Removal of external antenna fairings and base plates are required						
53-635-00-01	AWL	53-05-02-211-815	1.1	50000 FC	36000 FC	600 700 800	ALL
	Inspect (Detailed) the skin near the fastener locations around the antenna cutout, stringers, and antenna nutplates on both the left and right sides of the aircraft at STA 500 between stringers S-6 and S-7. See Doc D626A001-DTR, DTR check form 53-30-01-5 for alternative inspections. **AIRPLANE NOTE:** For the 737-600 GPS Antennas are located at Sta 482A + 5, LBL 5 and RBL 5. **ACCESS NOTE:** Removal of antenna is required.						
53-635-00-02	AWL	53-05-02-211-815	1.1	50000 FC	36000 FC	600 700 800	ALL
	Inspect (Detailed) the skin near the fastener locations around the antenna cutout, stringers, and antenna nutplates on both the left and right sides of the aircraft at STA 500 between stringers S-6 and S-7. See Doc D626A001-DTR, DTR check form 53-30-01-5 for alternative inspections. **AIRPLANE NOTE:** For the 737-600 GPS Antennas are located at Sta 482A + 5, LBL 5 and RBL 5. **ACCESS NOTE:** Removal of antenna is required.						
53-636-00-01	AWL	53-05-02-210-801	1.1	50000 FC	4000 FC	ALL	ALL
	Inspect (General Visual) the skin from STA 360 to STA 540 between stringers S-14 to S-17. See Doc D626A001-DTR, DTR check form 53-30-02-1 for alternative inspections. **ACCESS NOTE:** Remove or displace wing to body fairings as required to perform this inspection.						
53-636-00-02	AWL	53-05-02-210-801	1.1	50000 FC	4000 FC	ALL	ALL
	Inspect (General Visual) the skin from STA 360 to STA 540 between stringers S-14 to S-17. See Doc D626A001-DTR, DTR check form 53-30-02-1 for alternative inspections. **ACCESS NOTE:** Remove or displace wing to body fairings as required to perform this inspection.						
53-636-10-01	AWL	53-05-02-211-816	1.1	50000 FC	8000 FC	ALL	ALL
	Inspect (Detailed) the fuselage skin panels under the Wing to Body Fairing from STA 360 to STA 540. See Doc D626A001-DTR, DTR check form 53-30-02-4 for alternative inspections. **ACCESS NOTE:** Remove or displace wing to body fairings as required to perform this inspection.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-636-10-02	AWL	53-05-02-211-816	1.1	50000 FC	8000 FC	ALL	ALL
	Inspect (Detailed) the fuselage skin panels under the Wing to Body Fairing from STA 360 to STA 540. See Doc D626A001-DTR, DTR check form 53-30-02-4 for alternative inspections. **ACCESS NOTE:** Remove or displace wing to body fairings as required to perform this inspection.						
53-637-00-01	AWL	53-05-02-250-932	1.1	50000 FC	36000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (High Frequency Eddy Current) the upper skin along the upper fastener row at stringers S-4L and S-4R from STA 360 to STA 540. (PSE 53-30-04-1). See Doc. D626A001-DTR, DTR check form 53-10-03-1, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-41.						
53-637-00-02	AWL	53-05-02-250-932	1.1	50000 FC	36000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (High Frequency Eddy Current) the upper skin along the upper fastener row at stringers S-4L and S-4R from STA 360 to STA 540. (PSE 53-30-04-1). See Doc. D626A001-DTR, DTR check form 53-10-03-1, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-41.						
53-638-00-01	AWL	53-05-02-250-826	1.1	50000 FC	18000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (Low Frequency Eddy Current) the lower skin along the lower fastener row at stringers S-4L and S-4R from STA 360 to STA 540. See Doc. D626A001-DTR, DTR check form 53-30-04-2, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-50.						
53-638-00-02	AWL	53-05-02-250-826	1.1	50000 FC	18000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (Low Frequency Eddy Current) the lower skin along the lower fastener row at stringers S-4L and S-4R from STA 360 to STA 540. See Doc. D626A001-DTR, DTR check form 53-30-04-2, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-50.						
53-639-00-01	AWL	53-05-02-250-828	1.1	50000 FC	36000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (High Frequency Eddy Current) the upper skin along the upper fastener row at stringers S-10L and S-10R from STA 360 to STA 540. See Doc. D626A001-DTR, DTR check form 53-30-04-3, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-41.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-639-00-02	AWL	53-05-02-250-828	1.1	50000 FC	36000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (High Frequency Eddy Current) the upper skin along the upper fastener row at stringers S-10L and S-10R from STA 360 to STA 540. See Doc. D626A001-DTR, DTR check form 53-30-04-3, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-41.						
53-640-00-01	AWL	53-05-02-211-817	1.1	50000 FC	24000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (Detailed) the lower skin along the lower fastener row at stringers S-10L and S-10R from STA 360 to STA 540. See Doc. D626A001-DTR, DTR check form 53-30-04-4, for alternative inspections. **ACCESS NOTE:** Removal or displacement of interior sidewall panels and insulation blankets are required.						
53-640-00-02	AWL	53-05-02-211-817	1.1	50000 FC	24000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (Detailed) the lower skin along the lower fastener row at stringers S-10L and S-10R from STA 360 to STA 540. See Doc. D626A001-DTR, DTR check form 53-30-04-4, for alternative inspections. **ACCESS NOTE:** Removal or displacement of interior sidewall panels and insulation blankets are required.						
53-641-00-01	AWL	53-05-02-211-819	1.1	50000 FC	24000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (Detailed) the upper skin along the upper fastener row at stringers S-14L and S-14R from STA 360 to STA 540. See Doc. D626A001-DTR, DTR check form 53-30-04-5, for alternative inspections.						
53-641-00-02	AWL	53-05-02-211-819	1.1	50000 FC	24000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (Detailed) the upper skin along the upper fastener row at stringers S-14L and S-14R from STA 360 to STA 540. See Doc. D626A001-DTR, DTR check form 53-30-04-5, for alternative inspections.						
53-642-00-01	AWL	53-05-02-250-830	1.1	50000 FC	18000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (Low Frequency Eddy Current) the lower skin along the lower fastener row at stringers S-14L and S-14R from STA 360 to STA 540. See Doc. D626A001-DTR, DTR check form 53-30-04-6, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-50.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-642-00-02	AWL	53-05-02-250-830	1.1	50000 FC	18000 FC	600 700 700IGW 800 900 900ER	ALL

Inspect (Low Frequency Eddy Current) the lower skin along the lower fastener row at stringers S-14L and S-14R from STA 360 to STA 540.

See Doc. D626A001-DTR, DTR check form 53-30-04-6, for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-50.

53-643-00-01	AWL	53-05-02-250-832	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the upper (inner) skin along the upper fastener row at stringers S-24L and S-24R from STA 360 to STA 540, except at the cargo door cutout.

See Doc. D626A001-DTR, DTR check form 53-30-04-7, for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-50.

ACCESS NOTE: Remove Wing to Body Fairing as required to perform this inspection.

53-643-00-02	AWL	53-05-02-250-832	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the upper (inner) skin along the upper fastener row at stringers S-24L and S-24R from STA 360 to STA 540, except at the cargo door cutout.

See Doc. D626A001-DTR, DTR check form 53-30-04-7, for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-50.

ACCESS NOTE: Remove Wing to Body Fairing as required to perform this inspection.

53-644-00-01	AWL	53-05-02-211-821	1.1	50000 FC	4000 FC	ALL	ALL

Inspect (Detailed) the lower (outer) skin along the lower fastener row at stringers S-24L and S-24R from STA 360 to STA 540, except at the cargo door cutout.

See Doc. D626A001-DTR, DTR check form 53-30-04-8, for alternative inspections.

ACCESS NOTE: Remove Wing to Body Fairing as required to perform this inspection.

53-644-00-02	AWL	53-05-02-211-821	1.1	50000 FC	4000 FC	ALL	ALL

Inspect (Detailed) the lower (outer) skin along the lower fastener row at stringers S-24L and S-24R from STA 360 to STA 540, except at the cargo door cutout.

See Doc. D626A001-DTR, DTR check form 53-30-04-8, for alternative inspections.

ACCESS NOTE: Remove Wing to Body Fairing as required to perform this inspection.

53-645-00-01	AWL	53-05-02-211-863	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (Detailed) the window frames around each window from STA 360 to STA 540. (PSE 53-30-05).

See Doc D626A001-DTR, DTR check form 53-60-05-2 for alternative inspections.

ACCESS NOTE: Removal and/or displacement of passenger cabin sidewalls or sidewall window assemblies and insulation blankets is required.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-645-00-02	AWL	53-05-02-211-863	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (Detailed) the window frames around each window from STA 360 to STA 540. (PSE 53-30-05).
See Doc D626A001-DTR, DTR check form 53-60-05-2 for alternative inspections.

ACCESS NOTE: Removal and/or displacement of passenger cabin sidewalls or sidewall window assemblies and insulation blankets is required.

53-645-01-01	AWL	53-05-02-211-864	1.1	50000 FC	4000 FC	ALL	ALL

Inspect (Detailed) the window frames around each window from STA 360 to STA 540. (PSE 53-30-05).
See Doc D626A001-DTR, DTR check form 53-60-05-2 for alternative inspections.

53-645-01-02	AWL	53-05-02-211-864	1.1	50000 FC	4000 FC	ALL	ALL

Inspect (Detailed) the window frames around each window from STA 360 to STA 540. (PSE 53-30-05).
See Doc D626A001-DTR, DTR check form 53-60-05-2 for alternative inspections.

53-646-00-01	AWL	53-05-02-250-934	1.1	50000 FC	9000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the exposed edge of the bearstrap at both the forward and aft edge of the door at STA 440 and STA 492.4 from stringers S-18R to S-25R. (PSE 53-30-08).
See Doc D626A001-DTR, DTR check form 53-60-08-8 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-21.

53-646-10-01	AWL	53-05-02-250-833	1.1	50000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the outboard portion of the frame web for damage between stringers S-17R and S-26R, except at the door stops and sill locations.
See Doc D626A001-DTR, DTR check form 53-30-08-1 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-11-23.

ACCESS NOTE: Perform inspection with door open. Remove or displace cargo liners as required to perform this inspection.

53-646-20-01	AWL	53-05-02-130-801	1.1	50000 FC	18000 FC	ALL	ALL

Inspect (Ultrasonic) the outboard portion of the frame web for damage under all door stop fittings and sill clips.
See Doc D626A001-DTR, DTR check form 53-30-08-2 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 53-10-06.

ACCESS NOTE: Perform inspection with door open. Remove or displace cargo liners as required to perform this inspection.

53-646-30-01	AWL	53-05-02-250-834	1.1	50000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the forward and aft edge frame inner chords between stringers S-18R and S-26R.
See Doc D626A001-DTR, DTR check form 53-30-08-3 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-90.

ACCESS NOTE: Perform inspection with door open. Remove sealer at door stops as required.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-646-40-01	AWL	53-05-02-250-835	1.1	50000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the forward and aft edge frame inner chords between stringers S-17R and S-18R.
See Doc D626A001-DTR, DTR check form 53-30-08-4 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-87.

ACCESS NOTE: Remove cargo liners as required to perform the inspection.

53-646-50-01	AWL	53-05-02-250-836	1.1	50000 FC	18000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the forward and aft edge frame inner chords at stringer S-18R.
See Doc D626A001-DTR, DTR check form 53-30-08-5 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-80.

ACCESS NOTE: Remove cargo liners as required to perform the inspection.

53-646-60-01	AWL	53-05-02-250-837	1.1	50000 FC	9000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the bearstrap for two inches on each side of stringer S-24R at STA 440 and STA 492.4.
See Doc D626A001-DTR, DTR check form 53-30-08-6 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-91.

53-646-61-01	AWL	53-05-02-130-802	1.1	50000 FC	9000 FC	ALL	ALL

Inspect (Ultrasonic) the bearstrap for hidden damage under the stop backup fitting at stringer S-24R at STA 440 and STA 492.4.
See Doc D626A001-DTR, DTR check form 53-30-08-6 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 53-10-07.

53-646-62-01	AWL	53-05-02-211-865	1.1	34000 FC	18000 FC	ALL	ALL

Inspect (Detailed) the skin around the entire edge of the scuff plates at all four corners (upper/lower/fwd/aft) of the cargo door. (PSE 53-30-08-9).
See Doc. D626A001-DTR, DTR check form 53-60-08-9, for alternative inspections.

ACCESS NOTE: Forward cargo door must be open to perform this inspection. Scuff plate removal required.

53-646-63-01	AWL	53-05-02-210-808	1.1	34000 FC	18000 FC	ALL	ALL

Inspect (General Visual) the bearstrap at all four corners (upper/lower/fwd/aft) of the cargo door cutout. (PSE 53-30-08-9).
See Doc. D626A001-DTR, DTR check form 53-60-08-9, for alternative inspections.

ACCESS NOTE: Corner casting removal is required.

53-646-70-01	AWL	53-05-02-250-838	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the outer chord around the fasteners common to the chord and bearstrap.
See Doc. D626A001-DTR, DTR check form 53-30-08-10, for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-11-48.

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-646-71-01	AWL	53-05-02-250-839	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the bearstrap along the upper edge of the forward cargo door.

See Doc. D626A001-DTR, DTR check form 53-30-08-10, for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-11-48.

53-646-75-01	AWL	53-05-02-211-806	1.1	50000 FC	6000 FC	600 700 800 900 900ER	ALL

Inspect (Detailed) the upper sill inner chord. (PSE 53-30-08-11).

See Doc. D626A001-DTR, DTR check form 53-60-08-11, for alternative inspections.

53-646-80-01	AWL	53-05-02-250-840	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) around the fasteners common to the web at the lower main sill chords between STA 421 and 438 (for -600) and STA 461 and STA 478 (for -700/-800).

See Doc. D626A001-DTR, DTR check form 53-30-08-12, for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-33.

ACCESS NOTE: Removal of forward cargo door scuff plate is required to perform this inspection.

53-647-00-01	AWL	53-05-02-250-841	1.1	50000 FC	9000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the intercostal web for cracks adjacent to rivets and fastener holes (five locations at the forward and aft edge frames) common to the backup fitting and intercostal.

See Doc D626A001-DTR, DTR check form 53-30-09-3 for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-11-02.

ACCESS NOTE: Remove cargo liners as required.

53-648-00-01	AWL	53-05-02-250-842	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) around all fasteners in the angle (9 inches forward and 12 inches aft) of the wing to body intersection (STA 536) between STA 518 to STA 555, above stringer 24.

See Doc D626A001-DTR, DTR check form 53-30-11-1 for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-47.

ACCESS NOTE: Removal of wing to body fairings is required.

53-648-00-02	AWL	53-05-02-250-842	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) around all fasteners in the angle (9 inches forward and 12 inches aft) of the wing to body intersection (STA 536) between STA 518 to STA 555, above stringer 24.

See Doc D626A001-DTR, DTR check form 53-30-11-1 for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-47.

ACCESS NOTE: Removal of wing to body fairings is required.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-649-00-01	AWL	53-05-02-250-843	1.1	50000 FC	9000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) eight inches FWD and AFT of STA 536 along the edge of the inboard angle adjacent to the fuselage (FWD of STA 536) and the lower wing skin (AFT of STA 536).

See Doc D626A001-DTR, DTR check form 53-30-11-02 for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-47.

ACCESS NOTE: Removal of wing to body fairings, duct located aft of STA 536 and sealant along edge of angle to body contour and lower wing skin is required.

53-649-00-02	AWL	53-05-02-250-843	1.1	50000 FC	9000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) eight inches FWD and AFT of STA 536 along the edge of the inboard angle adjacent to the fuselage (FWD of STA 536) and the lower wing skin (AFT of STA 536).

See Doc D626A001-DTR, DTR check form 53-30-11-02 for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-47.

ACCESS NOTE: Removal of wing to body fairings, duct located aft of STA 536 and sealant along edge of angle to body contour and lower wing skin is required.

53-649-01-01	AWL	53-05-02-250-844	1.1	50000 FC	9000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) eight inches FWD and AFT of STA 536 along the inboard angle at the angle to fuselage interface (FWD of STA 536) and the angle to lower wing skin interface (AFT of STA 536).

See Doc D626A001-DTR, DTR check form 53-30-11-02 for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-47.

ACCESS NOTE: Removal of wing to body fairings, duct located aft of STA 536 and sealant along edge of angle to body contour and lower wing skin is required.

53-649-01-02	AWL	53-05-02-250-844	1.1	50000 FC	9000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) eight inches FWD and AFT of STA 536 along the inboard angle at the angle to fuselage interface (FWD of STA 536) and the angle to lower wing skin interface (AFT of STA 536).

See Doc D626A001-DTR, DTR check form 53-30-11-02 for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-47.

ACCESS NOTE: Removal of wing to body fairings, duct located aft of STA 536 and sealant along edge of angle to body contour and lower wing skin is required.

53-650-00-01	AWL	53-05-02-250-845	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) around the fasteners in the inboard and outboard angles eight inches forward and aft of STA 536.

See Doc. D626A001-DTR, DTR check form 53-30-11-03, for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-47.

ACCESS NOTE: Removal of wing to body fairings and duct located aft of STA 536 is required.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-650-00-02	AWL	53-05-02-250-845	1.1	50000 FC	36000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) around the fasteners in the inboard and outboard angles eight inches forward and aft of STA 536. See Doc. D626A001-DTR, DTR check form 53-30-11-03, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-47. **ACCESS NOTE:** Removal of wing to body fairings and duct located aft of STA 536 is required.						
53-652-00-01	AWL	53-05-02-211-866	1.1	50000 FC	4000 FC	ALL	ALL
	Inspect (Detailed) the skin around all of the fastener locations from stringer S-10L to S-10R, from STA 540 to STA 727, except at the lap splices and antennas. (53-40-01-1). See Doc D626A001-DTR, DTR check form 53-30-01-2 for alternative inspections.						
53-653-00-01	AWL	53-05-02-250-935	1.1	50000 FC	36000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (High Frequency Eddy Current) the upper skin along the upper fastener row at stringers S-4L and S-4R from STA 540 to STA 727. (PSE 53-40-03-1). See Doc. D626A001-DTR, DTR check form 53-10-03-1, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-41.						
53-653-00-02	AWL	53-05-02-250-935	1.1	50000 FC	36000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (High Frequency Eddy Current) the upper skin along the upper fastener row at stringers S-4L and S-4R from STA 540 to STA 727. (PSE 53-40-03-1). See Doc. D626A001-DTR, DTR check form 53-10-03-1, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-41.						
53-654-00-01	AWL	53-05-02-250-937	1.1	50000 FC	18000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (Low Frequency Eddy Current) the lower skin along the lower fastener row at stringers S-4L and S-4R from STA 540 to STA 727. (53-40-03-2). See Doc. D626A001-DTR, DTR check form 53-30-04-2, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-50.						
53-654-00-02	AWL	53-05-02-250-937	1.1	50000 FC	18000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (Low Frequency Eddy Current) the lower skin along the lower fastener row at stringers S-4L and S-4R from STA 540 to STA 727. (53-40-03-2). See Doc. D626A001-DTR, DTR check form 53-30-04-2, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-50.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-655-00-01	AWL	53-05-02-250-847	1.1	50000 FC	18000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (High Frequency Eddy Current) the upper skin along the upper fastener row at stringers S-10L and S-10R from STA 540 to STA 727. See Doc. D626A001-DTR, DTR check form 53-40-03-3, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-41.						
53-655-00-02	AWL	53-05-02-250-847	1.1	50000 FC	18000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (High Frequency Eddy Current) the upper skin along the upper fastener row at stringers S-10L and S-10R from STA 540 to STA 727. See Doc. D626A001-DTR, DTR check form 53-40-03-3, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-41.						
53-656-00-01	AWL	53-05-02-211-822	1.1	50000 FC	24000 FC	ALL	ALL
	Inspect (Detailed) stringers S-11 and S-13 from STA 540 to 727. See Doc. D626A001-DTR, DTR check form 53-40-04-1, for alternative inspections. **ACCESS NOTE:** Removal and/or displacement of passenger cabin sidewalls and insulation blankets is required.						
53-656-00-02	AWL	53-05-02-211-822	1.1	50000 FC	24000 FC	ALL	ALL
	Inspect (Detailed) stringers S-11 and S-13 from STA 540 to 727. See Doc. D626A001-DTR, DTR check form 53-40-04-1, for alternative inspections. **ACCESS NOTE:** Removal and/or displacement of passenger cabin sidewalls and insulation blankets is required.						
53-657-00-01	AWL	53-05-02-211-823	1.1	50000 FC	18000 FC	ALL	ALL
	Inspect (Detailed) the skin from stringers S-11 to S-13 between the windows from STA 540 to STA 727. See Doc. D626A001-DTR, DTR check form 53-40-04-2 for alternative inspections.						
53-657-00-02	AWL	53-05-02-211-823	1.1	50000 FC	18000 FC	ALL	ALL
	Inspect (Detailed) the skin from stringers S-11 to S-13 between the windows from STA 540 to STA 727. See Doc. D626A001-DTR, DTR check form 53-40-04-2 for alternative inspections.						
53-658-00-01	AWL	53-05-02-211-824	1.1	50000 FC	18000 FC	ALL	ALL
	Inspect (Detailed) stringers S-11 to S-13 at a distance of 10 inches forward and aft of STA 663. See Doc. D626A001-DTR, DTR check form 53-40-07-2 for alternative inspections. **ACCESS NOTE:** Remove and/or displace passenger cabin sidewall panels and insulation blankets.						
53-658-00-02	AWL	53-05-02-211-824	1.1	50000 FC	18000 FC	ALL	ALL
	Inspect (Detailed) stringers S-11 to S-13 at a distance of 10 inches forward and aft of STA 663. See Doc. D626A001-DTR, DTR check form 53-40-07-2 for alternative inspections. **ACCESS NOTE:** Remove and/or displace passenger cabin sidewall panels and insulation blankets.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-659-00-01	AWL	53-05-02-211-825	1.1	50000 FC	18000 FC	ALL	ALL

Inspect (Detailed) the keel beam side panel webs from STA 540 to STA 663.
See Doc. D626A001-DTR, DTR check form 53-40-08-1 for alternative inspections.

53-659-00-02	AWL	53-05-02-211-825	1.1	50000 FC	18000 FC	ALL	ALL

Inspect (Detailed) the keel beam side panel webs from STA 540 to STA 663.
See Doc. D626A001-DTR, DTR check form 53-40-08-1 for alternative inspections.

53-660-00-01	AWL	53-05-02-250-848	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the fastener row connecting the pressure bulkhead stiffener to the stiffener attachment fitting that joins the stiffener to the floor beam (at five locations) around the fastener/collar on the outboard side at LBL and RBL 45 and WL 202.6.
See Doc. D626A001-DTR, DTR check form 53-40-10-1 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-29.

ACCESS NOTE: Removal and/or displacement of aft cargo forward bulkhead, ceiling, sidewall panels and insulation blankets as required to perform the inspection.

53-660-00-02	AWL	53-05-02-250-848	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the fastener row connecting the pressure bulkhead stiffener to the stiffener attachment fitting that joins the stiffener to the floor beam (at five locations) around the fastener/collar on the outboard side at LBL and RBL 45 and WL 202.6.
See Doc. D626A001-DTR, DTR check form 53-40-10-1 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-29.

ACCESS NOTE: Removal and/or displacement of aft cargo forward bulkhead, ceiling, sidewall panels and insulation blankets as required to perform the inspection.

53-661-00-01	AWL	53-05-02-250-849	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the skin under the strap at stringer S-18 between the fasteners common to the strap and skin from STA 717 to STA 727.
See Doc. D626A001-DTR, DTR check form 53-40-11-1 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-49.

53-661-00-02	AWL	53-05-02-250-849	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the skin under the strap at stringer S-18 between the fasteners common to the strap and skin from STA 717 to STA 727.
See Doc. D626A001-DTR, DTR check form 53-40-11-1 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-49.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-662-00-01	AWL	53-05-02-250-850	1.1	50000 FC	36000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) around the bushings on each lug, three lugs per assembly, on the upper and lower surface at STA 663. See Doc. D626A001-DTR, DTR check form 53-40-12-1 for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-11-30.						
53-662-00-02	AWL	53-05-02-250-850	1.1	50000 FC	36000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) around the bushings on each lug, three lugs per assembly, on the upper and lower surface at STA 663. See Doc. D626A001-DTR, DTR check form 53-40-12-1 for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-11-30.						
53-663-00-01	AWL	53-05-02-250-851	1.1	50000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the frame inner chord at all fasteners common to the inner chord and to the inner splice plate between stringers S-8 and S-9. See Doc. D626A001-DTR, DTR check form 53-40-14-1 for alternative inspections. **ACCESS NOTE:** Removal and/or displacement of passenger cabin sidewalls and insulation blankets as required.						
53-663-00-02	AWL	53-05-02-250-851	1.1	50000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the frame inner chord at all fasteners common to the inner chord and to the inner splice plate between stringers S-8 and S-9. See Doc. D626A001-DTR, DTR check form 53-40-14-1 for alternative inspections. **ACCESS NOTE:** Removal and/or displacement of passenger cabin sidewalls and insulation blankets as required.						
53-664-00-01	AWL	53-05-02-211-826	1.1	50000 FC	18000 FC	ALL	ALL
	Inspect (Detailed) the skin panels at the outer chord from stringers S-9L to S-9R, on each side of splice 540, for cracks at the frame to skin fastener holes. See Doc. D626A001-DTR, DTR check form 53-40-14-4 for alternative inspections.						
53-665-00-01	AWL	53-05-02-211-827	1.1	50000 FC	18000 FC	ALL	ALL
	Inspect (Detailed) the skin on each side of STA 663, from stringers S-8 to S-11, on both the left and right sides. See Doc. D626A001-DTR, DTR check form 53-40-15-1 for alternative inspections.						
53-665-00-02	AWL	53-05-02-211-827	1.1	50000 FC	18000 FC	ALL	ALL
	Inspect (Detailed) the skin on each side of STA 663, from stringers S-8 to S-11, on both the left and right sides. See Doc. D626A001-DTR, DTR check form 53-40-15-1 for alternative inspections.						
53-666-00-01	AWL	53-05-02-250-852	1.1	50000 FC	24000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the bulkhead inner chord from stringers S-10 and S-17 on both the left and right hand sides. See Doc. D626A001-DTR, DTR check form 53-40-15-3 for alternative inspections. **ACCESS NOTE:** Removal and/or displacement of passenger cabin sidewalls and insulation blankets as required.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-666-00-02	AWL	53-05-02-250-852	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the bulkhead inner chord from stringers S-10 and S-17 on both the left and right hand sides.
See Doc. D626A001-DTR, DTR check form 53-40-15-3 for alternative inspections.

ACCESS NOTE: Removal and/or displacement of passenger cabin sidewalls and insulation blankets as required.

53-667-00-01	AWL	53-05-02-250-854	1.1	50000 FC	16000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the fail safe angle from inside the aft cargo bay at frame 727, from stringers S-21L to S-27L and stringers S-21R to S-27R.
See Doc. D626A001-DTR, DTR check form 53-40-16-1a for alternative inspections.

ACCESS NOTE: Removal and/or displacement of aft cargo forward bulkhead, ceiling, sidewall panels and insulation blankets as required to perform the inspection.

53-667-00-02	AWL	53-05-02-250-854	1.1	50000 FC	16000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the fail safe angle from inside the aft cargo bay at frame 727, from stringers S-21L to S-27L and stringers S-21R to S-27R.
See Doc. D626A001-DTR, DTR check form 53-40-16-1a for alternative inspections.

ACCESS NOTE: Removal and/or displacement of aft cargo forward bulkhead, ceiling, sidewall panels and insulation blankets as required to perform the inspection.

53-667-10-01	AWL	53-05-02-250-A98	1.1	50000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the chord at frame 727 from stringers S-21L to S-27L and stringers S-21R to S-27R.
See Doc. D626A001-DTR, DTR check form 53-40-16-1b for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-93.

53-667-10-02	AWL	53-05-02-250-A98	1.1	50000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the chord at frame 727 from stringers S-21L to S-27L and stringers S-21R to S-27R.
See Doc. D626A001-DTR, DTR check form 53-40-16-1b for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-93.

53-668-10-01	AWL	53-05-02-130-804	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) the frame around the six fasteners through the inner chord and web at STA 727 and WL 201.
See Doc. D626A001-DTR, DTR check form 53-40-16-2a for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 53-10-09.

AIRPLANE NOTE: This DTR Form is effective for all models from L/N 1193 and On.
See interval note for airplane specific threshold and repeat intervals.

INTERVAL NOTE: For 737-600 and -700 LN # 1193 and on - Threshold is 50,000 FC, Repeat is 24,000 FC.
For 737-700C/-700IGW/-800/-900 and -900ER - Threshold is 50,000 FC, Repeat is 24,000 FC.

ACCESS NOTE: Remove and/or displace aft cargo forward bulkhead, ceiling, sidewall panels and insulation blankets as required to perform the inspection.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-668-10-02	AWL	53-05-02-130-804	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) the frame around the six fasteners through the inner chord and web at STA 727 and WL 201.
See Doc. D626A001-DTR, DTR check form 53-40-16-2a for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 53-10-09.

AIRPLANE NOTE: This DTR Form is effective for all models from L/N 1193 and On.
See interval note for airplane specific threshold and repeat intervals.

INTERVAL NOTE: For 737-600 and -700 LN # 1193 and on - Threshold is 50,000 FC, Repeat is 24,000 FC.
For 737-700C/-700IGW/-800/-900 and -900ER - Threshold is 50,000 FC, Repeat is 24,000 FC.

ACCESS NOTE: Remove and/or displace aft cargo forward bulkhead, ceiling, sidewall panels and insulation blankets as required to perform the inspection.

53-669-00-01	AWL	53-05-02-250-853	1.1	50000 FC	36000 FC	600 700 700IGW 800 900 900ER	ALL

Inspect (High Frequency Eddy Current) the frame inner chord at STA 727 between stringers S-17 and S-21 on both the left and right hand sides.
See Doc. D626A001-DTR, DTR check form 53-40-16-3 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-11-25.

ACCESS NOTE: Remove and/or displace aft cargo forward bulkhead, ceiling, sidewall panels and insulation blankets as required to perform the inspection.

53-669-00-02	AWL	53-05-02-250-853	1.1	50000 FC	36000 FC	600 700 700IGW 800 900 900ER	ALL

Inspect (High Frequency Eddy Current) the frame inner chord at STA 727 between stringers S-17 and S-21 on both the left and right hand sides.
See Doc. D626A001-DTR, DTR check form 53-40-16-3 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-11-25.

ACCESS NOTE: Remove and/or displace aft cargo forward bulkhead, ceiling, sidewall panels and insulation blankets as required to perform the inspection.

53-670-00-01	AWL	53-05-02-211-832	1.1	50000 FC	9000 FC	ALL	ALL

Inspect (Detailed) the frame inner chord and web between stringers S-9L and S-9R at STA 727.
See Doc. D626A001-DTR, DTR check form 53-40-16-5 for alternative inspections.

ACCESS NOTE: Remove and/or displace passenger cabin ceiling panels and insulation as required to perform the inspection.

53-671-00-01	AWL	53-05-02-250-855	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the visible portion of the frame web above the splice angle on the forward side of the frame between stringers S-9 and S-10 on both sides of the aircraft at STA 727.
See Doc. D626A001-DTR, DTR check form 53-40-16-6 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-92.

ACCESS NOTE: Remove and/or displace passenger cabin ceiling panels and insulation as required to perform the inspection.

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-671-00-02	AWL	53-05-02-250-855	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the visible portion of the frame web above the splice angle on the forward side of the frame between stringers S-9 and S-10 on both sides of the aircraft at STA 727.
See Doc. D626A001-DTR, DTR check form 53-40-16-6 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-92.

ACCESS NOTE: Remove and/or displace passenger cabin ceiling panels and insulation as required to perform the inspection.

53-672-00-01	AWL	53-05-02-250-856	1.1	50000 FC	9000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the frame inner chord and fail-safe angle (around the fasteners common to the fail-safe angle), the forward frame web (around the fasteners common to the fail-safe angle), and the frame outer chord (around the fasteners common to the skin) between stringers S-10 and S-13.
See Doc. D626A001-DTR, DTR check form 53-40-17-1 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-11-19.

ACCESS NOTE: Remove and/or displace passenger cabin sidewalls and insulation blankets as required to perform the inspection.

53-672-00-02	AWL	53-05-02-250-856	1.1	50000 FC	9000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the frame inner chord and fail-safe angle (around the fasteners common to the fail-safe angle), the forward frame web (around the fasteners common to the fail-safe angle), and the frame outer chord (around the fasteners common to the skin) between stringers S-10 and S-13.
See Doc. D626A001-DTR, DTR check form 53-40-17-1 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-11-19.

ACCESS NOTE: Remove and/or displace passenger cabin sidewalls and insulation blankets as required to perform the inspection.

53-673-00-01	AWL	53-05-02-210-802	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (General Visual) the upper fastener through the web.
See Doc. D626A001-DTR, DTR check form 53-40-18-2 for alternative inspections.

ACCESS NOTE: Remove and/or displace passenger cabin sidewall air grilles, floor panels and insulation blankets as required to perform the inspection.

53-673-00-02	AWL	53-05-02-210-802	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (General Visual) the upper fastener through the web.
See Doc. D626A001-DTR, DTR check form 53-40-18-2 for alternative inspections.

ACCESS NOTE: Remove and/or displace passenger cabin sidewall air grilles, floor panels and insulation blankets as required to perform the inspection.

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-674-00-01	AWL	53-05-02-250-857	1.1	50000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the frame web around the fasteners common to the stringer clip at stringer S-16 and the forward and aft flanges of the frame inner chord from 6 inches above and below stringer S-16.

See Doc. D626A001-DTR, DTR check form 53-40-19-1 for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-40-05.

ACCESS NOTE: Remove and/or displace passenger cabin sidewall panels, sidewall air grilles, and insulation blankets as required to perform the inspection.

53-674-00-02	AWL	53-05-02-250-857	1.1	50000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the frame web around the fasteners common to the stringer clip at stringer S-16 and the forward and aft flanges of the frame inner chord from 6 inches above and below stringer S-16.

See Doc. D626A001-DTR, DTR check form 53-40-19-1 for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-40-05.

ACCESS NOTE: Remove and/or displace passenger cabin sidewall panels, sidewall air grilles, and insulation blankets as required to perform the inspection.

53-675-00-01	AWL	53-05-02-250-858	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the stub beam upper chord around the two fasteners common to the crease beam inner chord.

See Doc. D626A001-DTR, DTR check form 53-40-19-2 for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-14.

ACCESS NOTE: Remove and/or displace passenger cabin sidewall air grilles, floor panels and insulation blankets as required to perform the inspection.

53-675-00-02	AWL	53-05-02-250-858	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the stub beam upper chord around the two fasteners common to the crease beam inner chord.

See Doc. D626A001-DTR, DTR check form 53-40-19-2 for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-14.

ACCESS NOTE: Remove and/or displace passenger cabin sidewall air grilles, floor panels and insulation blankets as required to perform the inspection.

53-676-00-01	AWL	53-05-02-250-859	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the web of the stub beam around the fasteners common to the floor clip.

See Doc. D626A001-DTR, DTR check form 53-40-19-3 for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-09.

ACCESS NOTE: Remove and/or displace passenger cabin sidewall air grilles, floor panels and insulation blankets as required to perform the inspection.

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-676-00-02	AWL	53-05-02-250-859	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the web of the stub beam around the fasteners common to the floor clip.

See Doc. D626A001-DTR, DTR check form 53-40-19-3 for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-09.

ACCESS NOTE: Remove and/or displace passenger cabin sidewall air grilles, floor panels and insulation blankets as required to perform the inspection.

53-676-10-01	AWL	53-05-02-250-860	1.1	50000 FC	15000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the upper side of the upper flange on both the forward and aft sides from BL 45.5 to BL 64.6 on both the right and left sides.

See Doc. D626A001-DTR, DTR check form 53-40-19-3a for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-09.

ACCESS NOTE: Remove and/or displace passenger cabin sidewall air grilles, floor panels and insulation blankets as required to perform the inspection.

53-676-10-02	AWL	53-05-02-250-860	1.1	50000 FC	15000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the upper side of the upper flange on both the forward and aft sides from BL 45.5 to BL 64.6 on both the right and left sides.

See Doc. D626A001-DTR, DTR check form 53-40-19-3a for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-09.

ACCESS NOTE: Remove and/or displace passenger cabin sidewall air grilles, floor panels and insulation blankets as required to perform the inspection.

53-677-00-01	AWL	53-05-02-211-833	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (Detailed) the frame inner chord from stringers S-13 to S-15.

See Doc. D626A001-DTR, DTR check form 53-40-21-2 for alternative inspections.

ACCESS NOTE: Remove and/or displace passenger cabin sidewalls, sidewall air grilles, and insulation blankets as required to perform the inspection.

53-677-00-02	AWL	53-05-02-211-833	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (Detailed) the frame inner chord from stringers S-13 to S-15.

See Doc. D626A001-DTR, DTR check form 53-40-21-2 for alternative inspections.

ACCESS NOTE: Remove and/or displace passenger cabin sidewalls, sidewall air grilles, and insulation blankets as required to perform the inspection.

53-678-00-01	AWL	53-05-02-250-861	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the frame inner chord flange and around accessible fasteners common to the inner chord and stringer clips from stringers S-17 to S-14.

See Doc. D626A001-DTR, DTR check form 53-40-21-3 for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-17.

ACCESS NOTE: Remove and/or displace passenger cabin sidewalls, sidewall air grilles, and insulation blankets as required to perform the inspection.

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-678-00-02	AWL	53-05-02-250-861	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the frame inner chord flange and around accessible fasteners common to the inner chord and stringer clips from stringers S-17 to S-14.
See Doc. D626A001-DTR, DTR check form 53-40-21-3 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-17.

ACCESS NOTE: Remove and/or displace passenger cabin sidewalls, sidewall air grilles, and insulation blankets as required to perform the inspection.

53-679-00-01	AWL	53-05-02-250-862	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the stub beam upper chord from two inches inside the skin to a distance of twelve inches inboard and around any fasteners through the upper web and chord in this area.
See Doc. D626A001-DTR, DTR check form 53-40-21-4 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-06.

ACCESS NOTE: Remove and/or displace passenger cabin floor panels as required to perform the inspection.

53-679-00-02	AWL	53-05-02-250-862	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the stub beam upper chord from two inches inside the skin to a distance of twelve inches inboard and around any fasteners through the upper web and chord in this area.
See Doc. D626A001-DTR, DTR check form 53-40-21-4 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-06.

ACCESS NOTE: Remove and/or displace passenger cabin floor panels as required to perform the inspection.

53-682-00-01	AWL	53-05-02-250-867	1.1	50000 FC	4000 FC	800 900 900ER	ALL

Inspect (Low Frequency Eddy Current) the doublers around the fasteners common to the STA 578 cutout forward edge frame outer chord from stringers S-10 to S-13.
See Doc. D626A001-DTR, DTR check form 53-40-22-3 for alternative inspections.
NOTE: This inspection must be work in conjunction with either fatigue task 53-682-01 or 53-682-03 to meet DTR requirements..

AIRPLANE NOTE: Applicable to 737-800 (line number 9 and on) and 737-900 airplanes.

53-682-00-02	AWL	53-05-02-250-867	1.1	50000 FC	4000 FC	800 900 900ER	ALL

Inspect (Low Frequency Eddy Current) the doublers around the fasteners common to the STA 578 cutout forward edge frame outer chord from stringers S-10 to S-13.
See Doc. D626A001-DTR, DTR check form 53-40-22-3 for alternative inspections.
NOTE: This inspection must be work in conjunction with either fatigue task 53-682-01 or 53-682-03 to meet DTR requirements..

AIRPLANE NOTE: Applicable to 737-800 (line number 9 and on) and 737-900 airplanes.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-682-01-01	AWL	53-05-02-250-868	1.1	50000 FC	4000 FC	800 900 900ER	ALL

Inspect (High Frequency Eddy Current) the edges of the doublers from stringers S-11 to S-13.
See Doc. D626A001-DTR, DTR check form 53-40-22-3 for alternative inspections.
NOTE: This inspection may be optional. See DTR 53-40-22-3 for inspection requirements.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-15.

AIRPLANE NOTE: Applicable to 737-800 (line number 9 and on) and 737-900 airplanes.

ACCESS NOTE: Emergency Exit Door must be open to perform this inspection.

53-682-01-02	AWL	53-05-02-250-868	1.1	50000 FC	4000 FC	800 900 900ER	ALL

Inspect (High Frequency Eddy Current) the edges of the doublers from stringers S-11 to S-13.
See Doc. D626A001-DTR, DTR check form 53-40-22-3 for alternative inspections.
NOTE: This inspection may be optional. See DTR 53-40-22-3 for inspection requirements.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-15.

AIRPLANE NOTE: Applicable to 737-800 (line number 9 and on) and 737-900 airplanes.

ACCESS NOTE: Emergency Exit Door must be open to perform this inspection.

53-682-03-01	AWL	53-05-02-250-869	1.1	50000 FC	4000 FC	800 900 900ER	ALL

Inspect (High Frequency Eddy Current) the inner doubler from stringers S-11 to S-13.
See Doc. D626A001-DTR, DTR check form 53-40-22-3 for alternative inspections.
NOTE: This inspection may be optional. See DTR 53-40-22-3 for inspection requirements.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-15.

AIRPLANE NOTE: Applicable to 737-800 (line number 9 and on) and 737-900 airplanes.

ACCESS NOTE: Emergency Exit Door must be open to perform this inspection. Seal removal or displacement is required to perform the inspection.

53-682-03-02	AWL	53-05-02-250-869	1.1	50000 FC	4000 FC	800 900 900ER	ALL

Inspect (High Frequency Eddy Current) the inner doubler from stringers S-11 to S-13.
See Doc. D626A001-DTR, DTR check form 53-40-22-3 for alternative inspections.
NOTE: This inspection may be optional. See DTR 53-40-22-3 for inspection requirements.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-15.

AIRPLANE NOTE: Applicable to 737-800 (line number 9 and on) and 737-900 airplanes.

ACCESS NOTE: Emergency Exit Door must be open to perform this inspection. Seal removal or displacement is required to perform the inspection.

53-685-00-01	AWL	53-05-02-250-874	1.1	50000 FC	6000 FC	800 900 900ER	ALL

Inspect (High Frequency Eddy Current) the edges of the doublers, on the upper edge, at stringer S-11(from STA 578 to STA 601 and from STA 616 to STA 639).
See Doc. D626A001-DTR, DTR check form 53-40-22-6 for alternative inspections.
NOTE: Doors with external doublers at the upper forward corner refer to DTR 53-40-22-22 for area covered by the doubler.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-15.

ACCESS NOTE: Emergency Exit door must be open to perform the inspection.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-685-00-02	AWL	53-05-02-250-874	1.1	50000 FC	6000 FC	800 900 900ER	ALL

Inspect (High Frequency Eddy Current) the edges of the doublers, on the upper edge, at stringer S-11(from STA 578 to STA 601 and from STA 616 to STA 639).
See Doc. D626A001-DTR, DTR check form 53-40-22-6 for alternative inspections.
NOTE: Doors with external doublers at the upper forward corner refer to DTR 53-40-22-22 for area covered by the doubler.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-15.

ACCESS NOTE: Emergency Exit door must be open to perform the inspection.

53-685-01-01	AWL	53-05-02-250-875	1.1	50000 FC	6000 FC	800 900 900ER	ALL

Inspect (High Frequency Eddy Current) the inner doubler between the seal retainer and the frames and sills, on the upper edge, at stringer S-11 (from STA 578 to STA 601 and from STA 616 to STA 639).
See Doc. D626A001-DTR, DTR check form 53-40-22-6 for alternative inspections.
NOTE: Doors with external doublers at the upper forward corner refer to DTR 53-40-22-22 for area covered by the doubler.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-15.

ACCESS NOTE: Emergency Exit door must be open to perform this inspection. Remove or displace passenger cabin sidewall lining as required to perform this inspection.

53-685-01-02	AWL	53-05-02-250-875	1.1	50000 FC	6000 FC	800 900 900ER	ALL

Inspect (High Frequency Eddy Current) the inner doubler between the seal retainer and the frames and sills, on the upper edge, at stringer S-11 (from STA 578 to STA 601 and from STA 616 to STA 639).
See Doc. D626A001-DTR, DTR check form 53-40-22-6 for alternative inspections.
NOTE: Doors with external doublers at the upper forward corner refer to DTR 53-40-22-22 for area covered by the doubler.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-15.

ACCESS NOTE: Emergency Exit door must be open to perform this inspection. Remove or displace passenger cabin sidewall lining as required to perform this inspection.

53-686-00-01	AWL	53-05-02-130-805	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (Ultrasonic) the edge frames outer chord under the stop backup fittings at stringers S-11 and S-12.
See Doc. D626A001-DTR, DTR check form 53-40-22-7 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Section 53-10-10.

AIRPLANE NOTE: For the 737-600 and -700, STA 616 and STA 639.
For the 737-800, STA 578 and STA 639.

ACCESS NOTE: Emergency Exit door must be open to perform the inspection. Remove or displace passenger cabin sidewall lining as required to perform the inspection.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-686-00-02	AWL	53-05-02-130-805	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (Ultrasonic) the edge frames outer chord under the stop backup fittings at stringers S-11 and S-12.
See Doc. D626A001-DTR, DTR check form 53-40-22-7 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Section 53-10-10.

AIRPLANE NOTE: For the 737-600 and -700, STA 616 and STA 639.
For the 737-800, STA 578 and STA 639.

ACCESS NOTE: Emergency Exit door must be open to perform the inspection. Remove or displace passenger cabin sidewall lining as required to perform the inspection.

53-688-00-01	AWL	53-05-02-250-878	1.1	50000 FC	36000 FC	800 900 900ER	ALL

Inspect (High Frequency Eddy Current) the edges of the cutout doublers at all four lower corners (intersection of lower sill and edge frames) at stringer S-14 (from STA 578 to STA 607 and from STA 616 to STA 639).
See Doc. D626A001-DTR, DTR check form 53-40-22-9 for alternative inspections.
NOTE: Doors with external doublers at the lower door corner refer to DTR 53-40-22-22 for area covered by the doubler.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-15.

ACCESS NOTE: Emergency Exit door must be open to perform the inspection.

53-688-00-02	AWL	53-05-02-250-878	1.1	50000 FC	36000 FC	800 900 900ER	ALL

Inspect (High Frequency Eddy Current) the edges of the cutout doublers at all four lower corners (intersection of lower sill and edge frames) at stringer S-14 (from STA 578 to STA 607 and from STA 616 to STA 639).
See Doc. D626A001-DTR, DTR check form 53-40-22-9 for alternative inspections.
NOTE: Doors with external doublers at the lower door corner refer to DTR 53-40-22-22 for area covered by the doubler.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-15.

ACCESS NOTE: Emergency Exit door must be open to perform the inspection.

53-690-00-01	AWL	53-05-02-250-880	1.1	50000 FC	36000 FC	800 900 900ER	ALL

Inspect (Low Frequency Eddy Current) the doublers around the fasteners common to the edge frame at STA 578 (from stringers S-13 to S-15) and at STAs 601, 616 and 639 (from stringers S-10 to S-15).
See Doc. D626A001-DTR, DTR check form 53-40-22-11 for alternative inspections.
NOTE: Doors with external doublers at the lower door corners refer to DTR 53-40-22-22 for area covered by doubler.

53-690-00-02	AWL	53-05-02-250-880	1.1	50000 FC	36000 FC	800 900 900ER	ALL

Inspect (Low Frequency Eddy Current) the doublers around the fasteners common to the edge frame at STA 578 (from stringers S-13 to S-15) and at STAs 601, 616 and 639 (from stringers S-10 to S-15).
See Doc. D626A001-DTR, DTR check form 53-40-22-11 for alternative inspections.
NOTE: Doors with external doublers at the lower door corners refer to DTR 53-40-22-22 for area covered by doubler.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-692-00-01	AWL	53-05-02-210-803	1.1	50000 FC	18000 FC	800 900 900ER	ALL

Inspect (General Visual) the width and thickness of the lower sill inner splice strap and around the five fasteners at STAs 578, 601, 616 and 639.
Note: Fastener location is three FWD and two AFT at STAs 578/616 and two FWD and three AFT at STAs 601/639.
See Doc. D626A001-DTR, DTR check form 53-40-22-13 for alternative inspections.

ACCESS NOTE: Emergency Exit Door must be open to perform the inspection. Remove or displace passenger cabin sidewall lining as required to perform the inspection.

53-692-00-02	AWL	53-05-02-210-803	1.1	50000 FC	18000 FC	800 900 900ER	ALL

Inspect (General Visual) the width and thickness of the lower sill inner splice strap and around the five fasteners at STAs 578, 601, 616 and 639.
Note: Fastener location is three FWD and two AFT at STAs 578/616 and two FWD and three AFT at STAs 601/639.
See Doc. D626A001-DTR, DTR check form 53-40-22-13 for alternative inspections.

ACCESS NOTE: Emergency Exit Door must be open to perform the inspection. Remove or displace passenger cabin sidewall lining as required to perform the inspection.

53-695-00-01	AWL	53-05-02-211-835	1.1	50000 FC	9000 FC	ALL	ALL

Inspect (Detailed) the door stops attached to the forward and aft edge frames, six fittings per door.
See Doc. D626A001-DTR, DTR check form 53-40-23-1 for alternative inspections.

ACCESS NOTE: Emergency Exit Door must be open to perform the inspection.

53-695-00-02	AWL	53-05-02-211-835	1.1	50000 FC	9000 FC	ALL	ALL

Inspect (Detailed) the door stops attached to the forward and aft edge frames, six fittings per door.
See Doc. D626A001-DTR, DTR check form 53-40-23-1 for alternative inspections.

ACCESS NOTE: Emergency Exit Door must be open to perform the inspection.

53-696-00-01	AWL	53-05-02-211-836	1.1	50000 FC	36000 FC	800 900 900ER	ALL

Inspect (Detailed) the door stop intercostals (three locations) between STA 601 to STA 616.
See Doc. D626A001-DTR, DTR check form 53-40-23-2 for alternative inspections.

ACCESS NOTE: Emergency Exit Door must be open to perform this inspection. Removal of passenger cabin sidewall lining between Emergency Exit Doors is required to perform the inspection.

53-696-00-02	AWL	53-05-02-211-836	1.1	50000 FC	36000 FC	800 900 900ER	ALL

Inspect (Detailed) the door stop intercostals (three locations) between STA 601 to STA 616.
See Doc. D626A001-DTR, DTR check form 53-40-23-2 for alternative inspections.

ACCESS NOTE: Emergency Exit Door must be open to perform this inspection. Removal of passenger cabin sidewall lining between Emergency Exit Doors is required to perform the inspection.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-697-00-01	AWL	53-05-02-250-884	1.1	50000 FC	36000 FC	600 700 800 900 900ER	ALL

Inspect (High Frequency Eddy Current) the window frame edge inboard of the fasteners common to the door stop backup fitting attachment at stringers S-11 and S-12.

See Doc. D626A001-DTR, DTR check form 53-40-23-3 for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-24.

AIRPLANE NOTE: For the 737-600 and -700, STA 616.
For the 737-800, STA 578.

ACCESS NOTE: Removal of passenger cabin sidewall lining between Emergency Exit Doors is required to perform the inspection.

53-697-00-02	AWL	53-05-02-250-884	1.1	50000 FC	36000 FC	600 700 800 900 900ER	ALL

Inspect (High Frequency Eddy Current) the window frame edge inboard of the fasteners common to the door stop backup fitting attachment at stringers S-11 and S-12.

See Doc. D626A001-DTR, DTR check form 53-40-23-3 for alternative inspections.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-24.

AIRPLANE NOTE: For the 737-600 and -700, STA 616.
For the 737-800, STA 578.

ACCESS NOTE: Removal of passenger cabin sidewall lining between Emergency Exit Doors is required to perform the inspection.

53-697-10-01	AWL	53-05-02-210-837	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (General Visual) the angle between the rear spar extension and the pressure deck from the AFT or FWD side, including the bend radius.

Note: Either the AFT or FWD side inspection may be performed.

See Doc. D626A001-DTR, DTR check form 53-40-24-1 for alternative inspections.

53-698-00-01	AWL	53-05-02-211-867	1.1	50000 FC	4000 FC	ALL	ALL

Inspect (Detailed) the skin around all of the fastener locations from stringer S-10L to S-10R, from STA 727 to STA 887, except at the lap splices and antennas. (PSE 53-60-01-2).

See Doc D626A001-DTR, DTR check form 53-30-01-2 for alternative inspections.

ACCESS NOTE: Remove Dorsal Fin as required to perform the inspection.

53-699-00-01	AWL	53-05-02-211-837	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (Detailed) the crown skin panel at the ADF Antenna cutout (STA 727+9, RBL 5) and the SATCOM Antenna cutout (STA 747, stringer S-1).

See Doc. D626A001-DTR, DTR check form 53-60-01-4 for alternative inspections.

ACCESS NOTE: Removal of antenna, fairing and base plate as required to expose the skin to perform the inspection.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-700-00-01	AWL	53-05-02-250-885	1.1	50000 FC	36000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) the skin for hidden cracks between the adaptor plate and stringers S-1 and S-2L. See Doc. D626A001-DTR, DTR check form 53-60-01-5 for alternative inspections.						
	ACCESS NOTE: Remove or displace passenger cabin ceiling panels and air conditioning duct as required to perform the inspection.						
53-701-00-01	AWL	53-05-02-211-838	1.1	50000 FC	18000 FC	ALL	ALL
	Inspect (Detailed) the skin under the antenna adaptor plate from stringers S-1 to S-2L at STA 767. See Doc. D626A001-DTR, DTR check form 53-60-01-5 for alternative inspections.						
	ACCESS NOTE: Remove or displace passenger cabin ceiling panels and air conditioning duct as required to perform the inspection.						
53-702-00-01	AWL	53-05-02-250-886	1.1	50000 FC	36000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) the skin for hidden cracks between the adaptor plate and stringers S-1 and S-2L. See Doc. D626A001-DTR, DTR check form 53-60-01-6 for alternative inspections.						
	ACCESS NOTE: Remove or displace passenger cabin ceiling panels and air conditioning duct as required to perform the inspection.						
53-702-01-01	AWL	53-05-02-211-839	1.1	50000 FC	12000 FC	ALL	ALL
	Inspect (Detailed) the skin under the antenna adaptor plate from stringers S-1 to S-2L at STA 767. See Doc. D626A001-DTR, DTR check form 53-60-01-6 for alternative inspections.						
	ACCESS NOTE: Remove or displace passenger cabin ceiling panels and air conditioning duct as required to perform the inspection.						
53-703-00-01	AWL	53-05-02-210-805	1.1	50000 FC	4000 FC	ALL	ALL
	Inspect (General Visual) the skin from STA 727 to STA 887 between stringers S-14 to S-17. (PSE 53-60-02-1). See Doc D626A001-DTR, DTR check form 53-30-02-1 for alternative inspections.						
	ACCESS NOTE: Remove or displace wing to body fairings as required to perform the inspection.						
53-703-00-02	AWL	53-05-02-210-805	1.1	50000 FC	4000 FC	ALL	ALL
	Inspect (General Visual) the skin from STA 727 to STA 887 between stringers S-14 to S-17. (PSE 53-60-02-1). See Doc D626A001-DTR, DTR check form 53-30-02-1 for alternative inspections.						
	ACCESS NOTE: Remove or displace wing to body fairings as required to perform the inspection.						
53-703-10-01	AWL	53-05-02-211-868	1.1	50000 FC	8000 FC	ALL	ALL
	Inspect (Detailed) the fuselage skin panels under the Wing to Body Fairing from STA 727 to STA 887. (53-60-02-4). See Doc D626A001-DTR, DTR check form 53-30-02-4 for alternative inspections.						
	ACCESS NOTE: Remove or displace wing to body fairings as required to perform this inspection.						
53-703-10-02	AWL	53-05-02-211-868	1.1	50000 FC	8000 FC	ALL	ALL
	Inspect (Detailed) the fuselage skin panels under the Wing to Body Fairing from STA 727 to STA 887. (53-60-02-4). See Doc D626A001-DTR, DTR check form 53-30-02-4 for alternative inspections.						
	ACCESS NOTE: Remove or displace wing to body fairings as required to perform this inspection.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-704-00-01	AWL	53-05-02-250-940	1.1	50000 FC	36000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (High Frequency Eddy Current) the upper skin along the upper fastener row at stringers S-4L and S-4R from STA 727 to STA 887. (PSE 53-60-04-1). See Doc. D626A001-DTR, DTR check form 53-10-03-1, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-41.						
53-704-00-02	AWL	53-05-02-250-940	1.1	50000 FC	36000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (High Frequency Eddy Current) the upper skin along the upper fastener row at stringers S-4L and S-4R from STA 727 to STA 887. (PSE 53-60-04-1). See Doc. D626A001-DTR, DTR check form 53-10-03-1, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-41.						
53-705-00-01	AWL	53-05-02-250-942	1.1	50000 FC	18000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (Low Frequency Eddy Current) the lower skin along the lower fastener row at stringers S-4L and S-4R from STA 727 to STA 887. (PSE 53-60-04-2). See Doc. D626A001-DTR, DTR check form 53-30-04-2, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-50.						
53-705-00-02	AWL	53-05-02-250-942	1.1	50000 FC	18000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (Low Frequency Eddy Current) the lower skin along the lower fastener row at stringers S-4L and S-4R from STA 727 to STA 887. (PSE 53-60-04-2). See Doc. D626A001-DTR, DTR check form 53-30-04-2, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-50.						
53-706-00-01	AWL	53-05-02-250-944	1.1	50000 FC	36000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (High Frequency Eddy Current) the upper skin along the upper fastener row at stringers S-10L and S-10R from STA 727 to STA 887. (PSE 53-60-04-3) See Doc. D626A001-DTR, DTR check form 53-30-04-3, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-41.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-706-00-02	AWL	53-05-02-250-944	1.1	50000 FC	36000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (High Frequency Eddy Current) the upper skin along the upper fastener row at stringers S-10L and S-10R from STA 727 to STA 887. (PSE 53-60-04-3) See Doc. D626A001-DTR, DTR check form 53-30-04-3, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-41.						
53-707-00-01	AWL	53-05-02-211-869	1.1	50000 FC	24000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (Detailed) the lower skin along the lower fastener row at stringers S-10L and S-10R from STA 727 to STA 887. (PSE 53-60-04-4) See Doc. D626A001-DTR, DTR check form 53-30-04-4, for alternative inspections. **ACCESS NOTE:** Removal or displacement of interior sidewall panels and insulation blankets are required.						
53-707-00-02	AWL	53-05-02-211-869	1.1	50000 FC	24000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (Detailed) the lower skin along the lower fastener row at stringers S-10L and S-10R from STA 727 to STA 887. (PSE 53-60-04-4) See Doc. D626A001-DTR, DTR check form 53-30-04-4, for alternative inspections. **ACCESS NOTE:** Removal or displacement of interior sidewall panels and insulation blankets are required.						
53-708-00-01	AWL	53-05-02-211-871	1.1	50000 FC	24000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (Detailed) the upper skin along the upper fastener row at stringers S-14L and S-14R from STA 727 to STA 887. (PSE 53-60-04-5) See Doc. D626A001-DTR, DTR check form 53-30-04-5, for alternative inspections.						
53-708-00-02	AWL	53-05-02-211-871	1.1	50000 FC	24000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (Detailed) the upper skin along the upper fastener row at stringers S-14L and S-14R from STA 727 to STA 887. (PSE 53-60-04-5) See Doc. D626A001-DTR, DTR check form 53-30-04-5, for alternative inspections.						
53-709-00-01	AWL	53-05-02-250-946	1.1	50000 FC	18000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (Low Frequency Eddy Current) the lower skin along the lower fastener row at stringers S-14L and S-14R from STA 727 to STA 887. (PSE 53-60-04-6) See Doc. D626A001-DTR, DTR check form 53-30-04-6, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-50.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-709-00-02	AWL	53-05-02-250-946	1.1	50000 FC	18000 FC	600 700 700IGW 800 900 900ER	ALL
	Inspect (Low Frequency Eddy Current) the lower skin along the lower fastener row at stringers S-14L and S-14R from STA 727 to STA 887. (PSE 53-60-04-6) See Doc. D626A001-DTR, DTR check form 53-30-04-6, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-50.						
53-710-00-01	AWL	53-05-02-250-887	1.1	50000 FC	36000 FC	ALL	ALL
	Inspect ((Low Frequency Eddy Current) the upper (inner) skin along the upper fastener row at stringers S-23L and S-23R from STA 727 to STA 887, except at the cargo door cutout. See Doc. D626A001-DTR, DTR check form 53-60-04-7, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-30-50.						
53-710-00-02	AWL	53-05-02-250-887	1.1	50000 FC	36000 FC	ALL	ALL
	Inspect ((Low Frequency Eddy Current) the upper (inner) skin along the upper fastener row at stringers S-23L and S-23R from STA 727 to STA 887, except at the cargo door cutout. See Doc. D626A001-DTR, DTR check form 53-60-04-7, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-30-50.						
53-711-00-01	AWL	53-05-02-211-840	1.1	50000 FC	24000 FC	ALL	ALL
	Inspect (Detailed) the window frames around each window from STA 727 to STA 888. See Doc D626A001-DTR, DTR check form 53-60-05-2 for alternative inspections. **ACCESS NOTE:** Removal and/or displacement of passenger cabin sidewalls or sidewall window assemblies and insulation blankets as required.						
53-711-00-02	AWL	53-05-02-211-840	1.1	50000 FC	24000 FC	ALL	ALL
	Inspect (Detailed) the window frames around each window from STA 727 to STA 888. See Doc D626A001-DTR, DTR check form 53-60-05-2 for alternative inspections. **ACCESS NOTE:** Removal and/or displacement of passenger cabin sidewalls or sidewall window assemblies and insulation blankets as required.						
53-711-01-01	AWL	53-05-02-211-841	1.1	50000 FC	4000 FC	ALL	ALL
	Inspect (Detailed) the window frames around each window from STA 727 to STA 888. See Doc D626A001-DTR, DTR check form 53-60-05-2 for alternative inspections.						
53-711-01-02	AWL	53-05-02-211-841	1.1	50000 FC	4000 FC	ALL	ALL
	Inspect (Detailed) the window frames around each window from STA 727 to STA 888. See Doc D626A001-DTR, DTR check form 53-60-05-2 for alternative inspections.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-712-00-01	AWL	53-05-02-250-948	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) around the fasteners common to the web at the lower main sill chords between STA 807 and STA 827. (PSE 53-60-08).
See Doc. D626A001-DTR, DTR check form 53-30-08-12, for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-33.

ACCESS NOTE: Removal of aft cargo door scuff plate is required to perform the inspection.

53-713-00-01	AWL	53-05-02-250-949	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the upper sill outer chord around the fasteners common to the chord and bearstrap. (PSE 53-60-08).
See Doc. D626A001-DTR, DTR check form 53-30-08-10, for alternative inspections.

ACCESS NOTE: Remove or displace aft cargo door lining as required to perform the inspection.

53-713-01-01	AWL	53-05-02-250-974	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the bearstrap along the upper edge of the aft cargo door. (53-60-08).
See Doc. D626A001-DTR, DTR check form 53-30-08-10, for alternative inspections.

ACCESS NOTE: Remove or displace aft cargo door lining as required to perform the inspection.

53-714-00-01	AWL	53-05-02-250-950	1.1	50000 FC	9000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the bearstrap for two inches on each side of stringer S-24R at STA 794.4 and STA 847. (PSE 53-60-08)
See Doc D626A001-DTR, DTR check form 53-30-08-6 for alternative inspections.

53-714-01-01	AWL	53-05-02-130-809	1.1	50000 FC	9000 FC	ALL	ALL

Inspect (Ultrasonic) the bearstrap for hidden damage under the stop backup fitting at stringer S-24R at STA 794.4 and STA 847. (PSE 53-60-08)
See Doc D626A001-DTR, DTR check form 53-30-08-6 for alternative inspections.

53-715-00-01	AWL	53-05-02-250-888	1.1	50000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the frame inner chord between the web and failsafe strap from stringers S-18R to S-26R.
See Doc D626A001-DTR, DTR check form 53-60-08-1 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-88.

ACCESS NOTE: Cargo Door must be open to perform the inspection.

53-716-00-01	AWL	53-05-02-250-889	1.1	50000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the forward and aft edge frame inner chord fail-safe straps between stringers S-17R and S-18R.
See Doc D626A001-DTR, DTR check form 53-60-08-2 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-88.

ACCESS NOTE: Cargo Door must be open to perform the inspection. Remove or displace aft cargo sidewall and ceiling lining as required to perform the inspection.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-717-00-01	AWL	53-05-02-250-890	1.1	50000 FC	18000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) both the forward and aft edge frame inner chords at stringer S-18R.
See Doc D626A001-DTR, DTR check form 53-60-08-3 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-81.

ACCESS NOTE: Remove or displace aft cargo sidewall and ceiling lining as required to perform the inspection.

53-718-00-01	AWL	53-05-02-250-891	1.1	50000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the outboard portion of the web on the outer chord between stringers S-16R and S-26R at STA 794.37 and STA 847 (except at door stops and sills location).
See Doc D626A001-DTR, DTR check form 53-60-08-4 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-22.

ACCESS NOTE: Aft cargo door must be open to perform the inspection.

53-719-00-01	AWL	53-05-02-130-806	1.1	50000 FC	18000 FC	ALL	ALL

Inspect (Ultrasonic) the outboard portion of the frame web under the door stop fittings and sill clips at STA 794.37 and STA 847.
See Doc D626A001-DTR, DTR check form 53-60-08-5 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Section 53-10-08.

ACCESS NOTE: Aft cargo door must be open to perform the inspection. Remove or displace aft cargo lining as required to perform the inspection.

53-720-00-01	AWL	53-05-02-250-892	1.1	50000 FC	9000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the exposed edge of the bearstrap at both the forward and aft edge of the door at STA 794.4 and STA 847 from stringers S-18R to S-25R.
See Doc D626A001-DTR, DTR check form 53-60-08-8 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-21.

53-721-00-01	AWL	53-05-02-211-842	1.1	34000 FC	18000 FC	ALL	ALL

Inspect (Detailed) the skin at all four corners (upper/lower/FWD/AFT) of the cargo door cutout.
See Doc. D626A001-DTR, DTR check form 53-60-08-9, for alternative inspections.

ACCESS NOTE: Scuff plate removal required.

53-721-10-01	AWL	53-05-02-210-804	1.1	34000 FC	18000 FC	ALL	ALL

Inspect (General Visual) the bearstrap at all four corners (upper/lower/fwd/aft) of the cargo door cutout.
See Doc. D626A001-DTR, DTR check form 53-60-08-9, for alternative inspections.

ACCESS NOTE: Corner casting removal is required.

53-721-20-01	AWL	53-05-02-211-828	1.1	50000 FC	6000 FC	600 700 800 900 900ER	ALL

Inspect (DET) the upper sill inner chord.
See Doc. D626A001-DTR, DTR check form 53-60-08-11, for alternative inspections.

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-722-00-01	AWL	53-05-02-250-893	1.1	50000 FC	9000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the intercostal web for cracks adjacent to the rivet and fastener holes. Five door stops locations on both the forward and aft edge frames.
See Doc. D626A001-DTR, DTR check form 53-60-09-3, for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-82.

ACCESS NOTE: Aft cargo door must be open to perform the inspection. Remove or displace aft cargo sidewall lining as required to perform the inspection.

53-723-00-01	AWL	53-05-02-250-951	1.1	50000 FC	36000 FC	600 700 700IGW 800 900 900ER	ALL

Inspect (High Frequency Eddy Current) the upper skin along the upper fastener row at stringers S-4L and S-4R from STA 887 to STA 1016. (PSE 53-70-03-1).
See Doc. D626A001-DTR, DTR check form 53-10-03-1, for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-41.

53-723-00-02	AWL	53-05-02-250-951	1.1	50000 FC	36000 FC	600 700 700IGW 800 900 900ER	ALL

Inspect (High Frequency Eddy Current) the upper skin along the upper fastener row at stringers S-4L and S-4R from STA 887 to STA 1016. (PSE 53-70-03-1).
See Doc. D626A001-DTR, DTR check form 53-10-03-1, for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-41.

53-724-00-01	AWL	53-05-02-250-894	1.1	50000 FC	18000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the lower skin along the lower fastener row at stringers S-4L and S-4R from STA 887 to STA 1016.
See Doc. D626A001-DTR, DTR check form 53-70-03-2, for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-30-50.

53-724-00-02	AWL	53-05-02-250-894	1.1	50000 FC	18000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the lower skin along the lower fastener row at stringers S-4L and S-4R from STA 887 to STA 1016.
See Doc. D626A001-DTR, DTR check form 53-70-03-2, for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-30-50.

53-725-00-01	AWL	53-05-02-211-845	1.1	50000 FC	9000 FC	ALL	ALL

Inspect (Detailed) the upper skin along the upper fastener row at stringer S-14L (from STA 888 to STA 947, and from STA 1006 to STA 1016) and at stringer S-14R (from STA 888 to STA 947, and from STA 996 to STA 1016).
See Doc D626A001-DTR, DTR check form 53-70-03-3 for alternative inspections.

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-725-00-02	AWL	53-05-02-211-845	1.1	50000 FC	9000 FC	ALL	ALL

Inspect (Detailed) the upper skin along the upper fastener row at stringer S-14L (from STA 888 to STA 947, and from STA 1006 to STA 1016) and at stringer S-14R (from STA 888 to STA 947, and from STA 996 to STA 1016).
See Doc D626A001-DTR, DTR check form 53-70-03-3 for alternative inspections.

53-726-00-01	AWL	53-05-02-250-895	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the lower skin along the lower fastener row at stringer S-14L (from STA 888 to STA 947 and from STA 1006 to STA 1016) and at stringer S-14R (from STA 888 to STA 947 and from STA 996 to STA 1016).
See Doc D626A001-DTR, DTR check form 53-70-03-4 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-30-50.

53-726-00-02	AWL	53-05-02-250-895	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the lower skin along the lower fastener row at stringer S-14L (from STA 888 to STA 947 and from STA 1006 to STA 1016) and at stringer S-14R (from STA 888 to STA 947 and from STA 996 to STA 1016).
See Doc D626A001-DTR, DTR check form 53-70-03-4 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-30-50.

53-727-00-01	AWL	53-05-02-211-873	1.1	50000 FC	4000 FC	ALL	ALL

Inspect (Detailed) the skin around all of the fastener locations from stringer S-10L to S-10R, from STA 887 to STA 1016, except at the lap splices and antennas. (PSE 53-70-04-1).
See Doc D626A001-DTR, DTR check form 53-30-01-2 for alternative inspections.

ACCESS NOTE: Remove Dorsal Fin as required to perform the inspection.

53-728-00-01	AWL	53-05-02-250-896	1.1	50000 FC	9000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the four fastener locations at the #1, #2, #6 and #7 stop locations.
See Doc D626A001-DTR, DTR check form 53-70-07-3 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-35.

ACCESS NOTE: Remove or displace interior sidewall lining as required to perform the inspection.

53-729-00-01	AWL	53-05-02-250-897	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the door stop intercostals along the inner chord and around fasteners common to the web and doublers at the #1, #2, #6 and #7 stop locations.
See Doc D626A001-DTR, DTR check form 53-70-07-4 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-36.

ACCESS NOTE: Remove or displace interior sidewall and door lining as required to perform the inspection.

53-729-01-01	AWL	53-05-02-210-842	1.1	50000 FC	18000 FC	ALL	ALL

Inspect (General Visual) the door stop intercostals along the inner chord and around fasteners common to the web and doublers at the #1, #2, #6 and #7 stop locations.
See Doc D626A001-DTR, DTR check form 53-70-07-4 for alternative inspections.

ACCESS NOTE: Remove or displace interior sidewall and door lining as required to perform the inspection.

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-730-00-01	AWL	53-05-02-211-846	1.1	50000 FC	4000 FC	ALL	ALL
	Inspect (Detailed) the perimeter of the cutout and around the fasteners common to the edge frames and upper sill outer chords. See Doc D626A001-DTR, DTR check form 53-70-07-5 for alternative inspections. **ACCESS NOTE:** Remove or displace interior sidewall and door lining as required to perform the inspection.						
53-731-00-01	AWL	53-05-02-211-847	1.1	50000 FC	4000 FC	ALL	ALL
	Inspect (Detailed) the skin around the edges of the scuff plates. See Doc D626A001-DTR, DTR check form 53-70-07-6 for alternative inspections.						
53-731-01-01	AWL	53-05-02-250-898	1.1	50000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the skin around the fastener holes hidden by the scuff plate. See Doc D626A001-DTR, DTR check form 53-70-07-6 for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-30-34. **ACCESS NOTE:** Remove scuff plate.						
53-732-00-01	AWL	53-05-02-211-848	1.1	50000 FC	22000 FC	ALL	ALL
	Inspect (Detailed) the inner chord and web along the upper main sill from STA 951 to STA 1006. See Doc D626A001-DTR, DTR check form 53-70-07-11 for alternative inspections. **ACCESS NOTE:** Remove or displace passenger cabin sidewall and ceiling lining as required to perform the inspection.						
53-732-01-01	AWL	53-05-02-211-981	1.1	50000 FC	22000 FC	ALL	ALL
	Inspect (Detailed) the inner chord strap near the edge frames from STA 951 to STA 1006. See Doc D626A001-DTR, DTR check form 53-70-07-11 for alternative inspections. **ACCESS NOTE:** Remove or displace passenger cabin sidewall and ceiling lining as required to perform the inspection.						
53-732-10-01	AWL	53-05-02-250-997	1.1	50000 FC	36000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the first five fasteners on the upper flange of the lower main sill outer chord, aft of the edge frame. See Doc D626A001-DTR, DTR check form 53-70-07-12 for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-20. **ACCESS NOTE:** Remove scuff plates as required for access to the outer chord.						
53-733-00-01	AWL	53-05-02-211-874	1.1	50000 FC	22000 FC	ALL	ALL
	Inspect (Detailed) the inner chord and web along the upper main sill from STA 951 to STA 1006. (PSE 53-70-08). See Doc D626A001-DTR, DTR check form 53-70-07-11 for alternative inspections. **ACCESS NOTE:** Remove or displace passenger cabin sidewall and ceiling lining as required to perform the inspection.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-733-01-01	AWL	53-05-02-211-982	1.1	50000 FC	22000 FC	ALL	ALL

Inspect (Detailed) the inner chord strap near the edge frames from STA 951 to STA 1006. (PSE 53-70-08).
See Doc D626A001-DTR, DTR check form 53-70-07-11 for alternative inspections.

ACCESS NOTE: Remove or displace passenger cabin sidewall and ceiling lining as required to perform the inspection.

53-734-00-01	AWL	53-05-02-250-899	1.1	50000 FC	4000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the four fastener locations at the #1, #2, #5 and #6 stop locations.
See Doc D626A001-DTR, DTR check form 53-70-08-3 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-83.

ACCESS NOTE: Remove or displace passenger cabin sidewall and ceiling lining as required to perform the inspection.

53-734-01-01	AWL	53-05-02-250-900	1.1	50000 FC	12000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the four fastener locations at the #1, #2, #5 and #6 stop locations. For door stop #5, there are two locations in the strap hidden by the bracket.
See Doc D626A001-DTR, DTR check form 53-70-08-3 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-83.

ACCESS NOTE: Remove or displace passenger cabin sidewall and ceiling lining as required to perform the inspection.

53-735-00-01	AWL	53-05-02-250-901	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the door stop intercostals along the inner chord and around fasteners common to the web and doubler at the #1, #2, #5, and #6 stop locations.
See Doc D626A001-DTR, DTR check form 53-70-08-4 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-84.

ACCESS NOTE: Remove or displace passenger cabin sidewall and ceiling lining as required to perform the inspection.

53-735-01-01	AWL	53-05-02-210-843	1.1	50000 FC	18000 FC	ALL	ALL

Inspect (General Visual) the door stop intercostals along the inner chord and around fasteners common to the web and doubler at the #1, #2, #5, and #6 stop locations.
See Doc D626A001-DTR, DTR check form 53-70-08-4 for alternative inspections.

ACCESS NOTE: Remove or displace passenger cabin sidewall and ceiling lining as required to perform the inspection.

53-736-00-01	AWL	53-05-02-211-849	1.1	50000 FC	4000 FC	ALL	ALL

Inspect (Detailed) the perimeter of the cutout and around the fasteners common to the edge frames and upper sill outer chords.
See Doc D626A001-DTR, DTR check form 53-70-08-5 for alternative inspections.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-737-00-01	AWL	53-05-02-211-875	1.1	50000 FC	4000 FC	ALL	ALL
	Inspect (Detailed) the skin around the edges of the scuff plates. (PSE 53-70-08-6). See Doc D626A001-DTR, DTR check form 53-70-07-6 for alternative inspections.						
53-737-01-01	AWL	53-05-02-250-953	1.1	50000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the skin around the fastener holes hidden by the scuff plate. (PSE 53-70-08-6). See Doc D626A001-DTR, DTR check form 53-70-07-6 for alternative inspections. **ACCESS NOTE:** Remove scuff plate.						
53-737-10-01	AWL	53-05-02-250-A79	1.1	50000 FC	36000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the first five fasteners on the upper flange of the lower main sill outer chord, aft of the edge frame. (53-70-08-12). See Doc D626A001-DTR, DTR check form 53-70-07-12 for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-20. **ACCESS NOTE:** Remove scuff plates as required for access to the outer chord.						
53-738-00-01	AWL	53-05-02-211-876	1.1	50000 FC	24000 FC	ALL	ALL
	Inspect (Detailed) the window frames around each window from STA 888 to STA 927. (PSE 53-70-09) See Doc D626A001-DTR, DTR check form 53-60-05-2 for alternative inspections. **ACCESS NOTE:** Removal and/or displacement of passenger cabin sidewalls or sidewall window assemblies and insulation blankets as required.						
53-738-00-02	AWL	53-05-02-211-876	1.1	50000 FC	24000 FC	ALL	ALL
	Inspect (Detailed) the window frames around each window from STA 888 to STA 927. (PSE 53-70-09) See Doc D626A001-DTR, DTR check form 53-60-05-2 for alternative inspections. **ACCESS NOTE:** Removal and/or displacement of passenger cabin sidewalls or sidewall window assemblies and insulation blankets as required.						
53-738-01-01	AWL	53-05-02-211-877	1.1	50000 FC	4000 FC	ALL	ALL
	Inspect (Detailed) the window frames around each window from STA 888 to STA 927. (PSE 53-70-09) See Doc D626A001-DTR, DTR check form 53-60-05-2 for alternative inspections.						
53-738-01-02	AWL	53-05-02-211-877	1.1	50000 FC	4000 FC	ALL	ALL
	Inspect (Detailed) the window frames around each window from STA 888 to STA 927. (PSE 53-70-09) See Doc D626A001-DTR, DTR check form 53-60-05-2 for alternative inspections.						
53-739-00-01	AWL	53-05-02-211-850	1.1	50000 FC	9000 FC	ALL	ALL
	Inspect (Detailed) the pressure dome webs between the stiffeners and tear straps. See Doc D626A001-DTR, DTR check form 53-80-01-2 for alternative inspections. **AIRPLANE NOTE:** Applicable to all models except -800FPB and -900ER.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-740-00-01	AWL	53-05-02-211-851	1.1	50000 FC	9000 FC	ALL	ALL

Inspect (Detailed) the pressure dome web lap splices along the fastener rows adjacent to the radial stiffeners.
See Doc D626A001-DTR, DTR check form 53-80-01-3 for alternative inspections.

AIRPLANE NOTE: Applicable to all models except -800FPB and -900ER.

53-741-00-01	AWL	53-05-02-250-902	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the forward side of the pressure dome web along the aft fastener row attaching the web to the pressure chord. Inspect at the edge of each stiffener/clip and around the two fasteners on each side of the stiffener.
See Doc D626A001-DTR, DTR check form 53-80-01-4 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-11.

AIRPLANE NOTE: For L/N 721 and on, the inspection also includes the web adjacent to the stiffeners between stringers S-1 and S-3, S-3 and S-5, and S-5 and S-7 on both sides of the aircraft. Applicable to all models except -800FPB and -900ER.

ACCESS NOTE: Remove necessary passenger cabin and aft cargo interiors as required to perform the inspection.

53-742-00-01	AWL	53-05-02-250-903	1.1	50000 FC	12000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the pressure dome webs along the aft fastener row attaching the web to the pressure chord and between the stiffener locations from stringers S-5L to S-7L and S-5R to S-9R.
See Doc D626A001-DTR, DTR check form 53-80-01-5A for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-38.

AIRPLANE NOTE: Applicable to all models except -800FPB and -900ER.

ACCESS NOTE: Remove necessary passenger cabin interiors as required to perform the inspection.

53-743-00-01	AWL	53-05-02-250-904	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the pressure dome webs along the aft fastener row attaching the web to the pressure chord and between the stiffener locations outside of stringers S-5L to S-7L and S-5R to S-9R.
See Doc D626A001-DTR, DTR check form 53-80-01-5B for alternative inspections.

AIRPLANE NOTE: Applicable to all models except -800FPB and -900ER.

ACCESS NOTE: Remove necessary passenger cabin interiors as required to perform this inspection.

53-744-00-01	AWL	53-05-02-250-905	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) around the fasteners common to the pressure chord splices between stringers S-2 and S-3 and S-16 and S-17A.
See Doc D626A001-DTR, DTR check form 53-80-01-6 for alternative inspections.

AIRPLANE NOTE: For L/N 1057 and on, this inspection applies only to the splices between stringer S-16 and S-17A. Applicable to all models except -800FPB and -900ER.

ACCESS NOTE: Remove necessary passenger cabin interiors as required to perform the inspection.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-745-00-01	AWL	53-05-02-250-906	1.1	50000 FC	9000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the forward side of the pressure dome web around the fasteners common to the lap splice and the stiffeners.
See Doc D626A001-DTR, DTR check form 53-80-01-7 for alternative inspections.

AIRPLANE NOTE: Applicable to all models except -800FPB and -900ER.

ACCESS NOTE: Remove necessary passenger cabin interiors as required to perform the inspection.

53-746-00-01	AWL	53-05-02-250-907	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the forward side of the pressure dome web around the fasteners common to the tear strap.
See Doc D626A001-DTR, DTR check form 53-80-01-8 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-97.

AIRPLANE NOTE: Applicable to all models except -800FPB and -900ER.

ACCESS NOTE: Remove necessary passenger cabin interiors as required to perform the inspection.

53-747-00-01	AWL	53-05-02-250-908	1.1	50000 FC	9000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the aft side of the pressure dome web at the intersection of the tear straps and lap splice next to the stiffeners.
See Doc D626A001-DTR, DTR check form 53-80-01-9 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-08.

AIRPLANE NOTE: Applicable to all models except -800FPB and -900ER.

53-748-00-01	AWL	53-05-02-250-909	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the forward side of the pressure dome webs at the junction of the radial stiffeners/lap splices and the tear straps.
See Doc D626A001-DTR, DTR check form 53-80-01-11 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-07.

AIRPLANE NOTE: Applicable to all models except -800FPB and -900ER.

ACCESS NOTE: Remove necessary passenger cabin interiors as required to perform the inspection.

53-749-00-01	AWL	53-05-02-250-910	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the forward side of the pressure dome web around the fasteners common to the doubler, Y-Chord and the tear strap between stringers S-1 and S-3.
See Doc D626A001-DTR, DTR check form 53-80-01-13 for alternative inspections.

AIRPLANE NOTE: Applicable to all L/N 1057 an on, except -800FPB and -900ER.

ACCESS NOTE: Remove necessary passenger cabin interiors as required to perform the inspection.

53-749-10-01	AWL	53-05-02-211-888	1.1	36000 FC	10000 FC	ALL	ALL

Inspect (Detailed) the aft side of STA 1016 bulkhead web for oil cans.
Note: Refer to Structural Repair Manual, Section 53-80-08, for definition of oil can.
No DTR form available for PSE 53-80-01-14.

AIRPLANE NOTE: All aircraft L/N 1756 and on, except -800FPB and -900ER.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-750-00-01	AWL	53-05-02-211-852	1.1	50000 FC	33000 FC	ALL	ALL

Inspect (Detailed) the fittings on both sides of the bulkhead at STA 1016.
See Doc D626A001-DTR, DTR check form 53-80-02-1 for alternative inspections.

AIRPLANE NOTE: Applicable to all models except -800FPB and -900ER.

ACCESS NOTE: Remove necessary passenger cabin interiors as required to perform the inspection.

53-751-00-01	AWL	53-05-02-250-912	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the exposed forward and aft surfaces of the fitting lugs. Bolt removal is not required.
See Doc D626A001-DTR, DTR check form 53-80-02-2 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-78.

53-752-00-01	AWL	53-05-02-250-913	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the stringer splice fittings from stringer S-9L to S-9R at the first two fastener locations forward and aft of the STA 1016 bulkhead.
See Doc D626A001-DTR, DTR check form 53-80-03-1 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-79.

ACCESS NOTE: Remove or displace passenger cabin interior as required to perform the inspection.

53-753-00-01	AWL	53-05-02-250-914	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) both primary jackscrew fitting lugs on both sides around the bushing at the STA 1088 bulkhead.
See Doc D626A001-DTR, DTR check form 53-80-05-1 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-95.

ACCESS NOTE: Remove access panel as required. Remove/disconnect the jackscrew from the fitting and move aside for access to the lugs.

53-754-00-01	AWL	53-05-02-130-807	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (Ultrasonic) the top two fasteners in the outboard primary fittings common to the STA 1088 bulkhead.
See Doc D626A001-DTR, DTR check form 53-80-06-1 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 53-80-01.

ACCESS NOTE: Access fittings inside the tailcone on the forward and aft side of the STA 1088 Bulkhead. The top of the fittings are sandwiched between the splice angles and the bulkhead.

53-755-00-01	AWL	53-05-02-211-853	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (Detailed) the four primary fitting lugs, from the inside of the lugs, at STA 1088.
See Doc D626A001-DTR, DTR check form 53-80-06-2 for alternative inspections.

ACCESS NOTE: Remove vertical fin including primary and fail-safe bolts.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-756-00-01	AWL	53-05-02-250-915	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the edge of the outer chord and the web above and below the stab pivot line.
See Doc D626A001-DTR, DTR check form 53-80-07-1 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-85.

AIRPLANE NOTE: Applicable to L/N 1 to 1198.

53-757-00-01	AWL	53-05-02-250-916	1.1	50000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the web inboard along the failsafe strap from the top of the stabilizer attach fitting down 16 inches vertically.
See Doc D626A001-DTR, DTR check form 53-80-07-2 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 53-11-33.

AIRPLANE NOTE: Applicable to all aircrafts from L/N 1199 and on.

ACCESS NOTE: Enter aircraft through the tail cone access panel.

53-758-00-01	AWL	53-05-02-250-917	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the pivot fitting beams around the pivot pins at the STA 1156 hinge beam.
See Doc D626A001-DTR, DTR check form 53-80-08-1 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-86.

ACCESS NOTE: Remove sliding seals for access.

53-758-00-02	AWL	53-05-02-250-917	1.1	50000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the pivot fitting beams around the pivot pins at the STA 1156 hinge beam.
See Doc D626A001-DTR, DTR check form 53-80-08-1 for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 53-10-86.

ACCESS NOTE: Remove sliding seals for access.

53-759-00-01	AWL	53-05-02-230-801	1.1	50000 FC	14000 FC	ALL	ALL

Inspect (Dye Penetrant) both the inner and outer pivot pins at STA 1156.
See Doc D626A001-DTR, DTR check form 53-80-08-2 for alternative inspections.

ACCESS NOTE: Removal and separation of pivot pins is required to perform the inspection.

53-759-00-02	AWL	53-05-02-230-801	1.1	50000 FC	14000 FC	ALL	ALL

Inspect (Dye Penetrant) both the inner and outer pivot pins at STA 1156.
See Doc D626A001-DTR, DTR check form 53-80-08-2 for alternative inspections.

ACCESS NOTE: Removal and separation of pivot pins is required to perform the inspection.

53-760-00-01	AWL	53-05-02-211-854	1.1	50000 FC	18000 FC	ALL	ALL

Inspect (Detailed) the skin panels around the STA 1138 cutout.
See Doc D626A001-DTR, DTR check form 53-80-10-1 for alternative inspections.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-760-00-02	AWL	53-05-02-211-854	1.1	50000 FC	18000 FC	ALL	ALL
	Inspect (Detailed) the skin panels around the STA 1138 cutout. See Doc D626A001-DTR, DTR check form 53-80-10-1 for alternative inspections.						
53-800-00-01	MRB	05-41-01-210-801	1.1	120 DY	120 DY	ALL	ALL
	Perform an external zonal inspection (GV) of the lower fuselage. The nose landing gear wheel well and main landing gear wheel well are also included. Inspection is accomplished from the ground, without the use of stands or ladders. No additional access panels required. (EZAP) **INTERVAL NOTE:** The EZAP inspection requirement with interval 5500 FC/30 MO is satisfied by this zonal inspection.						
53-802-00-01	MRB	05-41-01-210-802	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the radome. **INTERVAL NOTE:** Whichever comes first.						
53-804-00-01	MRB	05-41-01-210-803	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the area forward of the nose wheel well - Section 41, Sta 178 to Sta 224.8. (EZAP) **INTERVAL NOTE:** Whichever comes first. This EZAP inspection requirement with interval 36000 FC/12 YR is satisfied by this zonal inspection.						
53-806-00-01	MRB	05-41-01-210-804	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the area above and outboard of the nose wheel well. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 36000 FC/12 YR is satisfied by this zonal inspection.						
53-808-00-01	MRB	05-41-01-210-805	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the electrical and electronics compartment. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 18000 FC/6 YR is satisfied by this zonal inspection. **ACCESS NOTE:** Forward airstair drip pan access panel must be removed if airstair installed. Access panel 117BL is only for airplanes with airstairs installed.						
53-810-00-01	MRB	05-41-01-210-806	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the electrical and electronics compartment access door. **INTERVAL NOTE:** Whichever comes first.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-812-00-01	MRB	05-41-01-210-807	1.1	120 DY	120 DY	ALL	ALL
	Perform an internal zonal inspection (GV) of the forward cargo compartment - section 43, STA 396 to forward cargo compartment aft bulkhead. **AIRPLANE NOTE:** For aircraft equipped with optional auxiliary fuel tanks, removal of tanks not required. **ACCESS NOTE:** No cargo liners removed.						
53-814-00-01	MRB	05-41-01-210-808	1.1 1.2	13200 FC 72 MO	13200 FC 72 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the area above the forward cargo compartment - section 43, STA 396 to forward cargo compartment aft bulkhead. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Ceiling panels removal required.						
53-816-00-01	MRB	05-41-01-210-809	1.1 1.2	36000 FC 12 YR	36000 FC 12 YR	ALL	ALL
	Perform an internal zonal inspection (GV) of the forward cargo compartment - section 43, STA 396 to forward cargo compartment aft bulkhead. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Sidewall panels removal required.						
53-818-00-01	MRB	05-41-01-210-810	1.1 1.2	2000 FC 240 DY	2000 FC 240 DY	ALL	ALL
	Perform an external zonal inspection (GV) of the forward cargo door surround structure fittings and stops - section 43, STA 460. **INTERVAL NOTE:** Whichever comes first.						
53-820-00-01	MRB	05-41-01-210-811	1.1 1.2	13200 FC 72 MO	13200 FC 72 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the area below the forward cargo compartment - Section 43, Sta 396 to forward cargo compartment aft bulkhead. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 18000 FC/6 YR is satisfied by this zonal inspection. **ACCESS NOTE:** Center floor panels removal required. Cargo loading system removed/displaced as required.						
53-822-00-01	MRB	05-41-01-210-812	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the air conditioning distribution bay - Section 43, forward cargo compartment bulkhead to Sta 540. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 36000 FC/12 YR is satisfied by this zonal inspection. **ACCESS NOTE:** Forward cargo compartment aft bulkhead center panel removal required.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-824-01-01	MRB	05-41-01-210-813	1.1 1.2	36000 FC 12 YR	36000 FC 12 YR	ALL	ALL

Perform an internal zonal inspection (GV) of the area above the center section wing box - section 44, STA 540 to STA 663.75.

Note: Access for this task is also provided when accomplishing tasks 53-878-00-01 and 53-880-00-01.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Floor panels removal required. Insulation as required.

53-826-00-01	MRB	05-41-01-210-814	1.1 1.2	13200 FC 72 MO	13200 FC 72 MO	ALL	ALL

Perform an internal zonal inspection (GV) of the pressure deck above the main landing gear wheel well - Section 44, Sta 663.75 to Sta 727. (EZAP)

INTERVAL NOTE: Whichever comes first. The EZAP inspection requirement with interval 36000 FC/12 YR is satisfied by this zonal inspection.

ACCESS NOTE: Floor panels removal required.

53-828-00-01	MRB	05-41-01-210-815	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL

Perform an internal zonal inspection (GV) of the keel beam (part) Sta 540 to 727 - Section 44. (EZAP)

INTERVAL NOTE: Whichever comes first. The EZAP inspection requirement with interval 36000 FC/12 YR is satisfied by this zonal inspection.

53-830-00-01	MRB	05-41-01-210-817	1.1	120 DY	120 DY	ALL	ALL

Perform an internal zonal inspection (GV) of the aft cargo compartment - section 46 and 47 (part), sta 727 to sta 947.5.

AIRPLANE NOTE: For aircraft equipped with optional fuel tanks, removal of tanks not required.

ACCESS NOTE: No cargo liners removed.

53-832-00-01	MRB	05-41-01-210-818	1.1 1.2	13200 FC 72 MO	13200 FC 72 MO	ALL	ALL

Perform an internal zonal inspection (GV) of the aft cargo compartment - section 46 and 47 (part), sta 727 to sta 947.5.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Upper angled sidewall and ceiling panels removal required.

53-834-00-01	MRB	05-41-01-210-819	1.1 1.2	36000 FC 12 YR	36000 FC 12 YR	ALL	ALL

Perform an internal zonal inspection (GV) of the aft cargo compartment - section 46 and 47 (part), sta 727 to sta 947.5.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Sidewall panels removal required.

53-836-00-01	MRB	05-41-01-210-820	1.1 1.2	2000 FC 240 DY	2000 FC 240 DY	ALL	ALL

Perform an external zonal inspection (GV) of the aft cargo door surround structure fittings and stops - section 46, sta 827.

INTERVAL NOTE: Whichever comes first.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-838-00-01	MRB	05-41-01-210-821	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL

Perform an internal zonal inspection (GV) of the aft cargo compartment vacuum waste compartment.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Vacuum waste compartment panels removal required.

53-840-00-01	MRB	05-41-01-210-822	1.1 1.2	13200 FC 72 MO	13200 FC 72 MO	ALL	ALL

Perform an internal zonal inspection (GV) of the area below the aft cargo compartment - Section 46 and 47 (part), Sta 727 to Sta 947.5. (EZAP)

INTERVAL NOTE: Whichever comes first. The EZAP inspection requirement with interval 18000 FC/6 YR is satisfied by this zonal inspection.

ACCESS NOTE: Center floor panels removal required. Cargo loading system removed/displaced as required.

53-842-00-01	MRB	05-41-01-210-823	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL

Perform an internal zonal inspection (GV) of the aft cargo compartment equipment bay - Section 47, Sta 947.5. to aft pressure bulkhead. (EZAP)

INTERVAL NOTE: Whichever comes first. The EZAP inspection requirement with interval 18000 FC/6 YR is satisfied by this zonal inspection.

ACCESS NOTE: Aft cargo panels at Sta 947 bulkhead removal required.

53-844-00-01	MRB	05-41-01-210-824	1.1	120 DY	120 DY	ALL	ALL

Perform an external zonal inspection (GV) of the wing to body fairing. Inspection is accomplished from the ground, without the use of stands or ladders. No additional access panels required.

53-846-00-01	MRB	05-41-01-210-825	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL

Perform an internal zonal inspection (GV) of the lower wing to body fairing - forward of wing box. (EZAP)

INTERVAL NOTE: Whichever comes first. The EZAP inspection requirement with interval 36000 FC/12 YR is satisfied by this zonal inspection.

ACCESS NOTE: Through access provided.

53-848-00-01	MRB	05-41-01-210-826	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL

Perform an internal zonal inspection (GV) of the lower wing to body fairing - under the wing box. (EZAP)

INTERVAL NOTE: Whichever comes first. The EZAP inspection requirement with interval 18000 FC/6 YR is satisfied by this zonal inspection.

ACCESS NOTE: Through access provided.

53-850-00-01	MRB	05-41-01-210-827	1.1 1.2	24000 FC 9 YR	24000 FC 8 YR	ALL	ALL

Perform an internal zonal inspection (GV) of the lower wing to body fairing - aft of wheel well. (EZAP)

INTERVAL NOTE: Whichever comes first. The EZAP inspection requirement with interval 36000 FC/12 YR is satisfied by this zonal inspection.

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-852-00-01	MRB	05-41-01-210-828	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the lower wing to body fairing - aft of wheel well. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Through access provided.						
53-854-02-01	MRB	05-41-01-210-829	1.1 1.2	24000 FC 9 YR	24000 FC 8 YR	ALL	ALL
	Perform an internal zonal inspection (GV) of the above wing, wing to body fairing - right side. **INTERVAL NOTE:** Whichever comes first.						
53-856-01-01	MRB	05-41-01-210-830	1.1 1.2	24000 FC 9 YR	24000 FC 8 YR	ALL	ALL
	Perform an internal zonal inspection (GV) of the above wing, wing to body fairing - left side. **INTERVAL NOTE:** Whichever comes first.						
53-858-00-01	MRB	05-41-02-210-801	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the flight control compartment - section 41, STA 178 to STA 270. **INTERVAL NOTE:** Whichever comes first.						
53-860-00-01	MRB	05-41-02-210-802	1.1 1.2	5500 FC 24 MO	5500 FC 24 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the flight control compartment - section 41, sta 178 to sta 270. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Control stand access panels 211A and 212A removal required.						
53-862-00-01	MRB	05-41-02-210-803	1.1 1.2	36000 FC 10 YR	36000 FC 10 YR	ALL	ALL
	Perform an internal zonal inspection (GV) of the flight control compartment - section 41, STA 178 to STA 270. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 36000 FC/12 YR is satisfied by this zonal inspection. **ACCESS NOTE:** With access provided. Seats removed. Control stand access panels, overhead and sidewall panels, and floor panel removal required.						
53-864-00-01	MRB	05-41-02-210-804	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the passenger compartment - aft of the control compartment to forward entry door - left and right - section 41, sta 270 to sta 360. **INTERVAL NOTE:** Whichever comes first.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-866-00-01	MRB	05-41-02-210-805	1.1 1.2	4800 FC 24 MO	4800 FC 24 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the passenger compartment - left and right - Section 41, STA 270 to STA 360. **INTERVAL NOTE:** Whichever comes first.						
53-868-00-01	MRB	05-41-02-210-806	1.1 1.2	36000 FC 9 YR	36000 FC 8 YR	ALL	ALL
	Perform an internal zonal inspection (GV) of the passenger compartment - aft of the control compartment to forward entry door - left and right - Section 41, Sta 270 to Sta 360. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 36000 FC/12 YR is satisfied by this zonal inspection. **ACCESS NOTE:** With access provided. Galleys and lavs removed. Floor panels, sidewall panels, and ceiling panels removal required.						
53-870-00-01	MRB	05-41-02-210-807	1.1 1.2	2000 FC 240 DY	2000 FC 240 DY	ALL	ALL
	Perform an external zonal inspection (GV) of the forward passenger entry door stops, latches and hinges - section 41, sta 345. **INTERVAL NOTE:** Whichever comes first.						
53-872-00-01	MRB	05-41-02-210-808	1.1 1.2	2000 FC 240 DY	2000 FC 240 DY	ALL	ALL
	Perform an external zonal inspection (GV) of the forward galley service door stops, latches, and hinges - section 41, sta 340. **INTERVAL NOTE:** Whichever comes first.						
53-874-00-01	MRB	05-41-02-210-810	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the forward passenger compartment - sta 360 to sta 663.75 - left and right - section 43 and 44 (part). **INTERVAL NOTE:** Whichever comes first.						
53-876-00-01	MRB	05-41-02-210-811	1.1 1.2	4800 FC 24 MO	4800 FC 24 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the forward passenger compartment - sta 360 to sta 663.75 - left and right - section 43 and 44 (part). **INTERVAL NOTE:** Whichever comes first.						
53-878-00-01	MRB	05-41-02-210-812	1.1 1.2	36000 FC 8 YR	36000 FC 8 YR	ALL	ALL
	Perform an internal zonal inspection (GV) of the forward passenger compartment - Sta 360 to Sta 663.75 wet areas (within approximately 20 inches from galley or lav) - left and right - Section 43 and 44 (part). (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 36000 FC/12 YR is satisfied by this zonal inspection. **ACCESS NOTE:** Galleys and lavs removed. Floor panels, sidewall panels, and ceiling panels removal required in areas where galleys and lavs are located.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-880-00-01	MRB	05-41-02-210-813	1.1 1.2	36000 FC 12 YR	36000 FC 12 YR	ALL	ALL

Perform an internal zonal inspection (GV) of the forward passenger compartment - Sta 360 to Sta 663.75 dry area (away from doors, galleys, and lavs) - left and right - Section 43 and 44 (part). (EZAP)

INTERVAL NOTE: Whichever comes first. The EZAP inspection requirement with interval 36000 FC/12 YR is satisfied by this zonal inspection.

ACCESS NOTE: Floor panels, sidewall panels, and ceiling panels removal required.

53-882-00-01	MRB	05-41-02-210-814	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL

Perform an external zonal inspection (gv) of the aft passenger compartment - sta 663.75 to the aft pressure bulkhead - left and right - section 44 (part), 46, and 47, sta 663.75 to the aft pressure bulkhead.

NOTE: Not applicable to airplanes with flat aft pressure bulkhead.

INTERVAL NOTE: Whichever comes first.

53-884-00-01	MRB	05-41-02-210-815	1.1 1.2	4800 FC 24 MO	4800 FC 24 MO	ALL	ALL

Perform an internal zonal inspection (GV) of the aft passenger compartment - STA 663.75 to the aft pressure bulkhead - left and right- section 44 (part), 46 and 47, STA 663.75 to aft pressure bulkhead.

NOTE: Not applicable to airplanes with flat aft pressure bulkhead.

INTERVAL NOTE: Whichever comes first.

53-886-00-01	MRB	05-41-02-210-816	1.1 1.2	36000 FC 8 YR	36000 FC 8 YR	ALL	ALL

Perform an internal zonal inspection (GV) of the aft passenger compartment - Sta 663.75 to the aft pressure bulkhead wet area (within approximately 20 inches from galley service door, passenger door, galley or lav) - left and right - Section 44 (part), 46, and 47, Sta 663.75 to aft pressure bulkhead. (EZAP)

NOTE: Not applicable to airplanes with flat aft pressure bulkhead.

INTERVAL NOTE: Whichever comes first. The EZAP inspection requirement with interval 36000 FC/12 YR is satisfied by this zonal inspection.

ACCESS NOTE: Galleys and lavs removed. Floor panels, sidewall panels, and ceiling panels removal required in areas where galleys and lavs are located.

53-888-00-01	MRB	05-41-02-210-817	1.1 1.2	36000 FC 12 YR	36000 FC 12 YR	ALL	ALL

Perform an internal zonal inspection (GV) of the aft passenger compartment - Sta 663.75 to the aft pressure bulkhead dry area (away from doors, galleys and lavs) - left and right - Section 44 (part), 46, and 47, Sta 663.75 to aft pressure bulkhead. (EZAP)

NOTE: Not applicable to airplanes with flat aft pressure bulkhead.

INTERVAL NOTE: Whichever comes first. The EZAP inspection requirement with interval 36000 FC/12 YR is satisfied by this zonal inspection.

ACCESS NOTE: Floor panels, sidewall panels, and ceiling panels removal required.

53-890-00-01	MRB	05-41-02-210-818	1.1 1.2	2000 FC 240 DY	2000 FC 240 DY	ALL	ALL

Perform an external zonal inspection (GV) of the aft passenger entry door stops, latches, and hinges - section 47, sta 980.

INTERVAL NOTE: Whichever comes first.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-892-00-01	MRB	05-41-02-210-819	1.1 1.2	2000 FC 240 DY	2000 FC 240 DY	ALL	ALL
	Perform an external zonal inspection (GV) of the aft galley service door stops, latches, and hinges - section 47, sta 980. **INTERVAL NOTE:** Whichever comes first.						
53-894-00-01	MRB	05-41-03-210-801	1.1	120 DY	120 DY	ALL	ALL
	Perform an external zonal inspection (GV) of the area aft of the pressure bulkhead. Inspection is accomplished from the ground, without the use of stands or ladders. No additional access panels required. NOTE: Not applicable to airplanes with flat aft pressure bulkhead.						
53-896-00-01	MRB	05-41-03-210-802	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the area aft of pressure bulkhead - Section 48, aft of aft pressure bulkhead to Sta 1088. (EZAP) NOTE: Not applicable to airplanes with flat aft pressure bulkhead. **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 36000 FC/12 YR is satisfied by this zonal inspection.						
53-898-00-01	MRB	05-41-03-210-803	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the stabilizer torsion box compartment - Section 48, Sta 1088 to Sta 1156. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 36000 FC/12 YR is satisfied by this zonal inspection.						
53-900-00-01	MRB	05-41-03-210-804	1.1 1.2	5500 FC 30 MO	5500 FC 30 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the APU compartment - Section 48, Sta 1088 to Sta 1156. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 5500 FC/30 MO is satisfied by this zonal inspection.						
53-902-00-01	MRB	05-41-03-210-805	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the tail cone - Section 48, Sta 1156 to Sta 1217. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 18000 FC/6 YR is satisfied by this zonal inspection.						
53-904-01-01	MRB	05-41-05-210-820	1.1 1.2	5500 FC 30 MO	5500 FC 30 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the flap support no. 4 - left wing - Section 46, Sta 727. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 5500 FC/30 MO is satisfied by this zonal inspection. **ACCESS NOTE:** Flaps extended.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
53-906-02-01	MRB	05-41-06-210-820	1.1 1.2	5500 FC 30 MO	5500 FC 30 MO	ALL	ALL

Perform an internal zonal inspection (GV) of the flap support no. 5 - right wing - Section 46, Sta 727. (EZAP)

INTERVAL NOTE: Whichever comes first. The EZAP inspection requirement with interval 5500 FC/30 MO is satisfied by this zonal inspection.

ACCESS NOTE: Flaps extended.

54-010-01-01	MRB	51-05-01-210-809 54-05-03-210-801	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL

Inspect left forward engine mount assembly, including fan case fitting, side links, hanger, and link pins.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Open fan cowl.

54-010-02-01	MRB	51-05-01-210-809 54-05-03-210-802	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL

Inspect right forward engine mount assembly, including fan case fitting, side links, hanger, and link pins.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Open fan cowl.

54-015-01-01	MRB	51-05-01-210-809 54-05-03-210-803	1.1 1.2	9 YR 18000 FC	9 YR 18000 FC	ALL	ALL

Inspect left strut attach bolts at forward engine mount.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Remove fan cowl.

54-015-02-01	MRB	51-05-01-210-809 54-05-03-210-804	1.1 1.2	9 YR 18000 FC	9 YR 18000 FC	ALL	ALL

Inspect right strut attach bolts at forward engine mount.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Remove fan cowl.

54-020-01-01	MRB	51-05-01-210-809 54-05-03-210-805	1.1	ENG CNG		ALL	ALL

Inspect forward and aft engine mount to strut shear pins.
Engine removal:
AMM 71-00-02 -POWER PLANT - REMOVAL/INSTALLATION
Related procedures:
AMM 71-21-01 - FORWARD ENGINE MOUNT REMOVAL/INSTALLATION
AMM 71-21-03 - AFT ENGINE MOUNT REMOVAL/INSTALLATION

INTERVAL NOTE: At engine removal.

ACCESS NOTE: Engine removal required.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
54-020-02-01	MRB	51-05-01-210-809 54-05-03-210-806	1.1	ENG CNG		ALL	ALL

Inspect forward and aft engine mount to strut shear pins.
Engine removal:
AMM 71-00-02 - POWER PLANT - REMOVAL/INSTALLATION
Related procedures:
AMM 71-21-01 - FORWARD ENGINE MOUNT REMOVAL/INSTALLATION
AMM 71-21-03 - AFT ENGINE MOUNT REMOVAL/INSTALLATION

INTERVAL NOTE: At engine removal.

ACCESS NOTE: Engine removal required.

54-030-01-01	MRB	51-05-01-210-809 54-05-03-210-807	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL

Inspect aft engine mount assembly, including thrust links and thrust link pins; mount to engine left, center and right links, including link pins; hanger and evener bar; attach bolts.

INTERVAL NOTE: Whichever comes first.

54-030-02-01	MRB	51-05-01-210-809 54-05-03-210-808	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL

Inspect aft engine mount assembly, including thrust links and thrust link pins; mount to engine left, center and right links, including link pins; hanger and evener bar; attach bolts.

INTERVAL NOTE: Whichever comes first.

54-040-01-01	MRB	51-05-01-210-809 54-05-03-210-809	1.1 1.2	48 MO 9000 FC	48 MO 9000 FC	ALL	ALL

Inspect strut to wing upper link, diagonal brace, side links, and strut attachment fittings.

INTERVAL NOTE: Whichever comes first.

54-040-02-01	MRB	51-05-01-210-809 54-05-03-210-810	1.1 1.2	48 MO 9000 FC	48 MO 9000 FC	ALL	ALL

Inspect strut to wing upper link, diagonal brace, side links, and strut attachment fittings.

INTERVAL NOTE: Whichever comes first.

54-050-01-01	MRB	51-05-01-210-809 54-05-03-211-801	1.1 1.2	48 MO 9000 FC	48 MO 9000 FC	ALL	ALL

Inspect pins and fuse pins on upper link, midspar, diagonal brace, and side links. Pin removal is not required.

INTERVAL NOTE: Whichever comes first.

54-050-02-01	MRB	51-05-01-210-809 54-05-03-211-802	1.1 1.2	48 MO 9000 FC	48 MO 9000 FC	ALL	ALL

Inspect pins and fuse pins on upper link, midspar, diagonal brace, and side links. Pin removal is not required.

INTERVAL NOTE: Whichever comes first.

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
54-060-01-01	MRB	51-05-01-210-809 54-05-03-211-803	1.1 1.2	10 YR 36000 FC	10 YR 36000 FC	ALL	ALL

Inspect the bores of pins and fuse pins on upper link, midspar, diagonal brace, and side links. Pin removal is not required.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: TR's must be open to remove access panels 431EL and 431ER.

54-060-02-01	MRB	51-05-01-210-809 54-05-03-211-804	1.1 1.2	10 YR 36000 FC	10 YR 36000 FC	ALL	ALL

Inspect the bores of pins and fuse pins on upper link, midspar, diagonal brace, and side links. Pin removal is not required.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: TR's must be open to remove access panels 441EL and 441ER.

54-070-01-01	MRB	51-05-01-210-803 54-05-03-210-811	1.1 1.2	9 YR 18000 FC	9 YR 18000 FC	ALL	ALL

Inspect external areas of strut box, including upper and lower spars, forward engine mount bulkhead, aft bulkhead, and side skins.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Remove MID and AFT insulation blanket/heat shields. Remove fan cowls. TR's must be open to remove access panels 431EL and 431ER. Engine removal not required.

54-070-02-01	MRB	51-05-01-210-803 54-05-03-210-812	1.1 1.2	9 YR 18000 FC	9 YR 18000 FC	ALL	ALL

Inspect external areas of strut box, including upper and lower spars, forward engine mount bulkhead, aft bulkhead, and side skins.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Remove MID and AFT insulation blanket/heat shields. Remove fan cowls. TR's must be open to remove access panels 441EL and 441ER. Engine removal not required.

54-080-01-01	MRB	51-05-01-210-803 54-05-03-210-813	1.1 1.2	9 YR 18000 FC	9 YR 18000 FC	ALL	ALL

Inspect internal areas of strut box, including upper and lower spars, forward and aft engine mount bulkheads, aft and mid bulkheads, and side skins.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Disassemble pneumatic ducts as required. TR's must be open to remove access panels 431EL and 431ER.

54-080-02-01	MRB	51-05-01-210-803 54-05-03-210-814	1.1 1.2	9 YR 18000 FC	9 YR 18000 FC	ALL	ALL

Inspect internal areas of strut box, including upper and lower spars, forward and aft engine mount bulkheads, aft and mid bulkheads, and side skins.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Disassemble pneumatic ducts as required. TR's must be open to remove access panels 441EL and 441ER.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
54-090-01-01	MRB	54-55-01-720-801 54-55-02-100-802	1.1 1.2	30 MO 9000 FC	30 MO 9000 FC	ALL	ALL

Functionally check the left engine forward strut and aft strut fairing drains.

INTERVAL NOTE: Whichever occurs first.

54-090-02-01	MRB	54-55-01-720-801 54-55-02-100-802	1.1 1.2	30 MO 9000 FC	30 MO 9000 FC	ALL	ALL

Functionally check the right engine forward strut and aft strut fairing drains.

INTERVAL NOTE: Whichever occurs first.

54-600-00-01	AWL	54-05-02-210-801	1.1	56000 FC	6000 FC	ALL	ALL

Inspect (General Visual) the lugs and clevises for all the links, fittings and pins.
See Doc D626A001-DTR, DTR check form 54-51-01, 54-51-02, 54-51-03, 54-51-04, 54-51-05, 54-51-06, 54-51-07, 54-51-08, 54-51-09, 54-51-15 for alternative inspections.

54-600-00-02	AWL	54-05-02-210-801	1.1	56000 FC	6000 FC	ALL	ALL

Inspect (General Visual) the lugs and clevises for all the links, fittings and pins.
See Doc D626A001-DTR, DTR check form 54-51-01, 54-51-02, 54-51-03, 54-51-04, 54-51-05, 54-51-06, 54-51-07, 54-51-08, 54-51-09, 54-51-15 for alternative inspections.

54-610-00-01	AWL	54-05-02-250-801	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) both legs of the lower spar chords between the forward and aft engine mounts: Nacelle STA 203.6 -209.9 left and right hand chords,
nacelle STA 212.3-222.0 left and right hand chords, nacelle STA 224.7-231.8 left hand chord, nacelle STA 234.4-240.4 left and right hand chords, nacelle STA 243.5-250.6 left and right hand chords.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-03.

ACCESS NOTE: Remove/displace heat shields and brackets as required.

54-610-00-02	AWL	54-05-02-250-801	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) both legs of the lower spar chords between the forward and aft engine mounts: Nacelle STA 203.6 -209.9 left and right hand chords,
nacelle STA 212.3-222.0 left and right hand chords, nacelle STA 224.7-231.8 left hand chord, nacelle STA 234.4-240.4 left and right hand chords, nacelle STA 243.5-250.6 left and right hand chords.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-03.

ACCESS NOTE: Remove/displace heat shields and brackets as required.

54-611-00-01	AWL	54-05-02-250-802	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the lower spar chord at the fire seal depressor bracket at nacelle STA 203.4 and nacelle STA 207.8 for both the left and right hand chords.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-03.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
54-611-00-02	AWL	54-05-02-250-802	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the lower spar chord at the fire seal depressor bracket at nacelle STA 203.4 and nacelle STA 207.8 for both the left and right hand chords.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-03.

54-611-01-01	AWL	54-05-02-130-801	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (Ultrasonic) the lower spar chord at the fire seal depressor bracket at nacelle STA 203.4 and at nacelle STA 207.8 on the left and right hand chords.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.

54-611-01-02	AWL	54-05-02-130-801	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (Ultrasonic) the lower spar chord at the fire seal depressor bracket at nacelle STA 203.4 and at nacelle STA 207.8 on the left and right hand chords.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.

54-612-00-01	AWL	54-05-02-130-802	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (Ultrasonic) the lower spar chord bracket at nacelle STA 216.0 on the left hand side and nacelle STA 218.0 on the right hand side.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative repeat inspection.

ACCESS NOTE: Removal of insulation heat shield is required. Remove fan cowls, thrust reversers, and engines as required.

54-612-00-02	AWL	54-05-02-130-802	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (Ultrasonic) the lower spar chord bracket at nacelle STA 216.0 on the left hand side and nacelle STA 218.0 on the right hand side.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative repeat inspection.

ACCESS NOTE: Removal of insulation heat shield is required. Remove fan cowls, thrust reversers, and engines as required.

54-613-00-01	AWL	54-05-02-130-803	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (Ultrasonic) the lower spar chord at the frames with brackets at nacelle STA 222.6 on the left and right chords and nacelle STA 242.7 on the left and right chords.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.

ACCESS NOTE: Remove fan cowls, thrust reversers, engine and insulation heat shields as required.

54-613-00-02	AWL	54-05-02-130-803	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (Ultrasonic) the lower spar chord at the frames with brackets at nacelle STA 222.6 on the left and right chords and nacelle STA 242.7 on the left and right chords.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.

ACCESS NOTE: Remove fan cowls, thrust reversers, engine and insulation heat shields as required.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
54-614-00-01	AWL	54-05-02-250-803	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the lower spar chord at the aft engine mount bulkhead.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative repeat inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-03.

54-614-00-02	AWL	54-05-02-250-803	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the lower spar chord at the aft engine mount bulkhead.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative repeat inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-03.

54-614-01-01	AWL	54-05-02-250-804	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the lower spar chord at the aft engine mount bulkhead.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-03.

ACCESS NOTE: Inspection requires the removal of engine mount.

54-614-01-02	AWL	54-05-02-250-804	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the lower spar chord at the aft engine mount bulkhead.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-03.

ACCESS NOTE: Inspection requires the removal of engine mount.

54-615-00-01	AWL	54-05-02-250-805	1.1	56000 FC	9000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) only the lower right spar web from nacelle STA 224.7 to nacelle STA 231.8.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-03.

ACCESS NOTE: Open/Remove Thrust Reverser as Required. Removal of Pneumatic Ducting Required.

54-615-00-02	AWL	54-05-02-250-805	1.1	56000 FC	9000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) only the lower right spar web from nacelle STA 224.7 to nacelle STA 231.8.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-03.

ACCESS NOTE: Open/Remove Thrust Reverser as Required. Removal of Pneumatic Ducting Required.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
54-616-00-01	AWL	54-05-02-250-806	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) only the lower right spar chord from nacelle STA 224.7 to nacelle STA 231.8.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-03.

ACCESS NOTE: Remove thrust reverser as required. Removal of insulation heat shields is required.

54-616-00-02	AWL	54-05-02-250-806	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) only the lower right spar chord from nacelle STA 224.7 to nacelle STA 231.8.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-03.

ACCESS NOTE: Remove thrust reverser as required. Removal of insulation heat shields is required.

54-617-00-01	AWL	54-05-02-250-807	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the horizontal leg of the left and right hand chords, common to the compression pad bracket at nacelle STA 244.9.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-03.

ACCESS NOTE: Remove thrust reverser as required. Removal of insulation heat shields is required.

54-617-00-02	AWL	54-05-02-250-807	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the horizontal leg of the left and right hand chords, common to the compression pad bracket at nacelle STA 244.9.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-03.

ACCESS NOTE: Remove thrust reverser as required. Removal of insulation heat shields is required.

54-617-01-01	AWL	54-05-02-130-804	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (Ultrasonic) the horizontal leg of the left and right hand chords, common to the compression pad bracket at nacelle STA 244.9.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.

ACCESS NOTE: Remove thrust reverser as required. Removal of insulation heat shield is required.

54-617-01-02	AWL	54-05-02-130-804	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (Ultrasonic) the horizontal leg of the left and right hand chords, common to the compression pad bracket at nacelle STA 244.9.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.

ACCESS NOTE: Remove thrust reverser as required. Removal of insulation heat shield is required.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
54-618-00-01	AWL	54-05-02-250-808	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the vertical leg of the compression pad bracket on the lower left and right spar chords at nacelle STA 244.9.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-03.

54-618-00-02	AWL	54-05-02-250-808	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the vertical leg of the compression pad bracket on the lower left and right spar chords at nacelle STA 244.9.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-03.

54-619-00-01	AWL	54-05-02-130-805	1.1	56000 FC	9000 FC	ALL	ALL

Inspect (Ultrasonic) the internal side of the vertical leg at nacelle STA 209.0 - 212.3, nacelle STA 231.8 - 234.4 on the left and right hand chords.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.

54-619-00-02	AWL	54-05-02-130-805	1.1	56000 FC	9000 FC	ALL	ALL

Inspect (Ultrasonic) the internal side of the vertical leg at nacelle STA 209.0 - 212.3, nacelle STA 231.8 - 234.4 on the left and right hand chords.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.

54-619-01-01	AWL	54-05-02-250-809	1.1	56000 FC	9000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the external side of the vertical leg at nacelle STA 209.0 - 212.3, and nacelle STA 231.8 - 234.4 on the left and right hand chords.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-03.

ACCESS NOTE: Remove thrust reversers as required.

54-619-01-02	AWL	54-05-02-250-809	1.1	56000 FC	9000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the external side of the vertical leg at nacelle STA 209.0 - 212.3, and nacelle STA 231.8 - 234.4 on the left and right hand chords.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-03.

ACCESS NOTE: Remove thrust reversers as required.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
54-620-00-01	AWL	54-05-02-250-810	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the horizontal leg on the left and right chords from nacelle STA 209.0 to nacelle STA 212.3 and from nacelle STA 231.8 to nacelle STA 234.4.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-03.

ACCESS NOTE: Remove thrust reversers as required.

54-620-00-02	AWL	54-05-02-250-810	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the horizontal leg on the left and right chords from nacelle STA 209.0 to nacelle STA 212.3 and from nacelle STA 231.8 to nacelle STA 234.4.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-03.

ACCESS NOTE: Remove thrust reversers as required.

54-621-00-01	AWL	54-05-02-211-801	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (Detailed) the lower spar chord aft of the aft engine mount bulkhead.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.

ACCESS NOTE: Remove thrust reversers as required. Removal of insulation heat shields required.

54-621-00-02	AWL	54-05-02-211-801	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (Detailed) the lower spar chord aft of the aft engine mount bulkhead.
See Doc. D626A001-DTR, DTR check form 54-51-10, for alternative inspections.

ACCESS NOTE: Remove thrust reversers as required. Removal of insulation heat shields required.

54-622-00-01	AWL	54-05-02-250-811	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the end pad bolt holes (4 locations) at the FWD engine mount bulkhead.
See Doc. D626A001-DTR, DTR check form 54-51-11, for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-03.

ACCESS NOTE: Bathtub fitting bolts must be sequentially (one at a time) removed for bolt hole eddy current inspection. Retorque tension bolts per dwg requirements.

54-622-00-02	AWL	54-05-02-250-811	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the end pad bolt holes (4 locations) at the FWD engine mount bulkhead.
See Doc. D626A001-DTR, DTR check form 54-51-11, for alternative inspections.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-03.

ACCESS NOTE: Bathtub fitting bolts must be sequentially (one at a time) removed for bolt hole eddy current inspection. Retorque tension bolts per dwg requirements.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
54-623-00-01	AWL	54-05-02-250-812	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the FWD and Aft flanges of both tension fittings common to the R1 fitting (4) attachment bolts. See Doc. D626A001-DTR, DTR check form 54-51-14, for alternative inspections. **ACCESS NOTE:** Internal access required.						
54-623-00-02	AWL	54-05-02-250-812	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the FWD and Aft flanges of both tension fittings common to the R1 fitting (4) attachment bolts. See Doc. D626A001-DTR, DTR check form 54-51-14, for alternative inspections. **ACCESS NOTE:** Internal access required.						
54-624-00-01	AWL	54-05-02-250-813	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) all exposed side skin surfaces within 4.5 inches of cutout at nacelle STA 252 and nacelle STA 270 left and right hand sides. See Doc. D626A001-DTR, DTR check form 54-51-16, for alternative inspections. **ACCESS NOTE:** Remove thrust reversers as required.						
54-624-00-02	AWL	54-05-02-250-813	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) all exposed side skin surfaces within 4.5 inches of cutout at nacelle STA 252 and nacelle STA 270 left and right hand sides. See Doc. D626A001-DTR, DTR check form 54-51-16, for alternative inspections. **ACCESS NOTE:** Remove thrust reversers as required.						
54-625-00-01	AWL	54-05-02-130-806	1.1	56000 FC	36000 FC	ALL	ALL
	Inspect (Ultrasonic) the strut side skin at nacelle STA 222.6, left and right sides and nacelle STA 242.7, left and right sides. See Doc. D626A001-DTR, DTR check form 54-51-16, for alternative inspections.						
54-625-00-02	AWL	54-05-02-130-806	1.1	56000 FC	36000 FC	ALL	ALL
	Inspect (Ultrasonic) the strut side skin at nacelle STA 222.6, left and right sides and nacelle STA 242.7, left and right sides. See Doc. D626A001-DTR, DTR check form 54-51-16, for alternative inspections.						
54-626-00-01	AWL	54-05-02-250-814	1.1	56000 FC	9000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the upper spar chord between the forward and aft engine mounts: Nacelle STA 200.9 - 211.5 on the left chords, nacelle STA 213.6 - 225.2 on the left chords, nacelle STA 226.1 - 233.6 on the left chords, nacelle STA 236.3 - 241.8 on the left and right chords. See Doc. D626A001-DTR, DTR check form 54-51-17, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-07.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
54-626-00-02	AWL	54-05-02-250-814	1.1	56000 FC	9000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the upper spar chord between the forward and aft engine mounts: Nacelle STA 200.9 - 211.5 on the left chords, nacelle STA 213.6 - 225.2 on the left chords, nacelle STA 226.1 - 233.6 on the left chords, nacelle STA 236.3 - 241.8 on the left and right chords. See Doc. D626A001-DTR, DTR check form 54-51-17, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-07.						
54-627-00-01	AWL	54-05-02-250-815	1.1	56000 FC	9000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the upper spar chord near the cutouts: Nacelle STA 200.9 - nacelle STA 211.5 on the right chord, nacelle STA 213.6 - nacelle STA 225.2 on the right chord, nacelle STA 226.1 - nacelle STA 233.6 on the right chord. See Doc. D626A001-DTR, DTR check form 54-51-17, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-07.						
54-627-00-02	AWL	54-05-02-250-815	1.1	56000 FC	9000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the upper spar chord near the cutouts: Nacelle STA 200.9 - nacelle STA 211.5 on the right chord, nacelle STA 213.6 - nacelle STA 225.2 on the right chord, nacelle STA 226.1 - nacelle STA 233.6 on the right chord. See Doc. D626A001-DTR, DTR check form 54-51-17, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-07.						
54-627-01-01	AWL	54-05-02-250-816	1.1	56000 FC	9000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the upper spar chord near the cutouts: Nacelle STA 200.9 - nacelle STA 211.5 on the right chord, nacelle STA 213.6 - nacelle STA 225.2 on the right chord, nacelle STA 226.1 - Nacelle STA 233.6 on the right chord. See Doc. D626A001-DTR, DTR check form 54-51-17, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-07.						
54-627-01-02	AWL	54-05-02-250-816	1.1	56000 FC	9000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the upper spar chord near the cutouts: Nacelle STA 200.9 - nacelle STA 211.5 on the right chord, nacelle STA 213.6 - nacelle STA 225.2 on the right chord, nacelle STA 226.1 - Nacelle STA 233.6 on the right chord. See Doc. D626A001-DTR, DTR check form 54-51-17, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-07.						
54-628-00-01	AWL	54-05-02-250-817	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the upper spar, R1 fitting and upper spar web on the (horizontal) chord only near the cutouts: Nacelle STA 200.9 to nacelle STA 211.5, nacelle STA 213.6 to nacelle STA 225.2, nacelle STA 226.1 to nacelle STA 233.6. See Doc. D626A001-DTR, DTR check form 54-51-17, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-07.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
54-628-00-02	AWL	54-05-02-250-817	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the upper spar, R1 fitting and upper spar web on the (horizontal) chord only near the cutouts: Nacelle STA 200.9 to nacelle STA 211.5, nacelle STA 213.6 to nacelle STA 225.2, nacelle STA 226.1 to nacelle STA 233.6. See Doc. D626A001-DTR, DTR check form 54-51-17, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-07.						
54-628-01-01	AWL	54-05-02-250-818	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the upper spar, R1 fitting and upper spar web on the (vertical) chord only near the cutouts: Nacelle STA 200.9 to nacelle STA 211.5, nacelle STA 213.6 to nacelle STA 225.2, nacelle STA 226.1 to nacelle STA 233.6. See Doc. D626A001-DTR, DTR check form 54-51-17, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-07.						
54-628-01-02	AWL	54-05-02-250-818	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the upper spar, R1 fitting and upper spar web on the (vertical) chord only near the cutouts: Nacelle STA 200.9 to nacelle STA 211.5, nacelle STA 213.6 to nacelle STA 225.2, nacelle STA 226.1 to nacelle STA 233.6. See Doc. D626A001-DTR, DTR check form 54-51-17, for alternative inspections. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 54-40-07.						
54-629-00-01	AWL	54-05-02-250-819	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the upper spar, R1 fitting and upper spar joints, web only near cutouts: Nacelle STA 200.9 - to nacelle STA 211.5, nacelle STA 213.6 - to nacelle STA 225.2, nacelle STA 226.1 - to nacelle STA 233.6. See Doc. D626A001-DTR, DTR check form 54-51-17, for alternative inspections.						
54-629-00-02	AWL	54-05-02-250-819	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the upper spar, R1 fitting and upper spar joints, web only near cutouts: Nacelle STA 200.9 - to nacelle STA 211.5, nacelle STA 213.6 - to nacelle STA 225.2, nacelle STA 226.1 - to nacelle STA 233.6. See Doc. D626A001-DTR, DTR check form 54-51-17, for alternative inspections.						
54-630-00-01	AWL	54-05-02-250-820	1.1	56000 FC	9000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) the chords, skins, webs, and bulkhead in all splices. Inspect the inside and the outside of the strut, and all structure buried in the splices using low frequency subsurface eddy current. See Doc. D626A001-DTR, DTR check form 54-51-17, for alternative inspections.						
54-630-00-02	AWL	54-05-02-250-820	1.1	56000 FC	9000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) the chords, skins, webs, and bulkhead in all splices. Inspect the inside and the outside of the strut, and all structure buried in the splices using low frequency subsurface eddy current. See Doc. D626A001-DTR, DTR check form 54-51-17, for alternative inspections.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
54-630-01-01	AWL	54-05-02-250-821	1.1	56000 FC	9000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the chords, skins, webs, and bulkhead in all splices. Inspect the inside and the outside of the strut, and all visible structure in the splices using high frequency surface eddy current. See Doc. D626A001-DTR, DTR check form 54-51-17, for alternative inspections.						
54-630-01-02	AWL	54-05-02-250-821	1.1	56000 FC	9000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the chords, skins, webs, and bulkhead in all splices. Inspect the inside and the outside of the strut, and all visible structure in the splices using high frequency surface eddy current. See Doc. D626A001-DTR, DTR check form 54-51-17, for alternative inspections.						
54-631-00-01	AWL	54-05-02-250-822	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the left and right upper spar chords at nacelle STA 222.6 and nacelle STA 210.6. See Doc. D626A001-DTR, DTR check form 54-51-17, for alternative inspections.						
54-631-00-02	AWL	54-05-02-250-822	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the left and right upper spar chords at nacelle STA 222.6 and nacelle STA 210.6. See Doc. D626A001-DTR, DTR check form 54-51-17, for alternative inspections.						
54-632-00-01	AWL	54-05-02-130-807	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (Ultrasonic) the hidden portion of the upper spar chord at nacelle STA 242.7 on the left and right chords. See Doc. D626A001-DTR, DTR check form 54-51-17, for alternative inspections.						
54-632-00-02	AWL	54-05-02-130-807	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (Ultrasonic) the hidden portion of the upper spar chord at nacelle STA 242.7 on the left and right chords. See Doc. D626A001-DTR, DTR check form 54-51-17, for alternative inspections.						
54-632-01-01	AWL	54-05-02-250-823	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the visible portion of the upper spar chord at nacelle STA 242.7 on the left and right chords. See Doc. D626A001-DTR, DTR check form 54-51-17, for alternative inspections.						
54-632-01-02	AWL	54-05-02-250-823	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the visible portion of the upper spar chord at nacelle STA 242.7 on the left and right chords. See Doc. D626A001-DTR, DTR check form 54-51-17, for alternative inspections.						
54-633-00-01	AWL	54-05-02-250-824	1.1	56000 FC	9000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) all exposed surfaces of the R3/R4 first fastener row on the vertical leg right and left sides. See Doc. D626A001-DTR, DTR check form 54-51-17, for alternative inspections.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
54-633-00-02	AWL	54-05-02-250-824	1.1	56000 FC	9000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) all exposed surfaces of the R3/R4 first fastener row on the vertical leg right and left sides. See Doc. D626A001-DTR, DTR check form 54-51-17, for alternative inspections.						
54-634-00-01	AWL	54-05-02-230-801	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (Penetrant) the entire forward engine mount hanger. The critical detail is the bolt hole detail at the top of the mount. See Doc. D626A001-DTR, DTR check form 54-55-03, for alternative inspections. **ACCESS NOTE:** Removal of engine and engine mount is required.						
54-634-00-02	AWL	54-05-02-230-801	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (Penetrant) the entire forward engine mount hanger. The critical detail is the bolt hole detail at the top of the mount. See Doc. D626A001-DTR, DTR check form 54-55-03, for alternative inspections. **ACCESS NOTE:** Removal of engine and engine mount is required.						
54-635-00-01	AWL	54-05-02-210-802	1.1	56000 FC	6000 FC	ALL	ALL
	Inspect (General Visual) the thrust link and the thrust link clevis lug. Lead crack is the failed thrust link. Critical detail is the intact thrust link clevis lug. See Doc. D626A001-DTR, DTR check form 54-55-05, for alternative inspections.						
54-635-00-02	AWL	54-05-02-210-802	1.1	56000 FC	6000 FC	ALL	ALL
	Inspect (General Visual) the thrust link and the thrust link clevis lug. Lead crack is the failed thrust link. Critical detail is the intact thrust link clevis lug. See Doc. D626A001-DTR, DTR check form 54-55-05, for alternative inspections.						
54-636-00-01	AWL	54-05-02-210-803	1.1	56000 FC	6000 FC	ALL	ALL
	Inspect (General Visual) the thrust link pin. Lead crack is the failed thrust link pin. Critical detail is the intact thrust link clevis lug. See Doc. D626A001-DTR, DTR check form 54-55-06, for alternative inspections.						
54-636-00-02	AWL	54-05-02-210-803	1.1	56000 FC	6000 FC	ALL	ALL
	Inspect (General Visual) the thrust link pin. Lead crack is the failed thrust link pin. Critical detail is the intact thrust link clevis lug. See Doc. D626A001-DTR, DTR check form 54-55-06, for alternative inspections.						
54-637-00-01	AWL	54-05-02-230-802	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (Penetrant) the entire aft engine mount assembly. The aft mount critical detail is the shear pin hole. See Doc. D626A001-DTR, DTR check form 54-55-10, for alternative inspections. **ACCESS NOTE:** The inspection requires the removal of the engine and disassembly of the engine mount.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
54-637-00-02	AWL	54-05-02-230-802	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (Penetrant) the entire aft engine mount assembly. The aft mount critical detail is the shear pin hole. See Doc. D626A001-DTR, DTR check form 54-55-10, for alternative inspections. **ACCESS NOTE:** The inspection requires the removal of the engine and disassembly of the engine mount.						
54-638-00-01	AWL	54-05-02-210-804	1.1	56000 FC	6000 FC	ALL	ALL
	Inspect (General Visual) the evener bar assembly outboard lugs. Lead crack is the failed evener bar (outboard lug). Critical detail is the intact thrust link clevis lug. See Doc. D626A001-DTR, DTR check form 54-55-11, for alternative inspections. **ACCESS NOTE:** The inspection requires the removal of the engine and disassembly of the engine mount.						
54-638-00-02	AWL	54-05-02-210-804	1.1	56000 FC	6000 FC	ALL	ALL
	Inspect (General Visual) the evener bar assembly outboard lugs. Lead crack is the failed evener bar (outboard lug). Critical detail is the intact thrust link clevis lug. See Doc. D626A001-DTR, DTR check form 54-55-11, for alternative inspections. **ACCESS NOTE:** The inspection requires the removal of the engine and disassembly of the engine mount.						
54-639-00-01	AWL	54-05-02-230-803	1.1	56000 FC	75000 FC	ALL	ALL
	Inspect (Penetrant) the entire aft engine mount evener bar. See Doc. D626A001-DTR, DTR check form 54-55-11, for alternative inspections. **ACCESS NOTE:** Inspection requires the removal and thorough cleaning of the evener bar.						
54-639-00-02	AWL	54-05-02-230-803	1.1	56000 FC	75000 FC	ALL	ALL
	Inspect (Penetrant) the entire aft engine mount evener bar. See Doc. D626A001-DTR, DTR check form 54-55-11, for alternative inspections. **ACCESS NOTE:** Inspection requires the removal and thorough cleaning of the evener bar.						
54-640-00-01	AWL	54-05-02-700-801	1.1	56000 FC	6000 FC	ALL	ALL
	Verify (Torque Check) all strut attach bolts on the forward and aft mounts. See Doc. D626A001-DTR, DTR check form 54-55-13, for alternative inspections.						
54-640-00-02	AWL	54-05-02-700-801	1.1	56000 FC	6000 FC	ALL	ALL
	Verify (Torque Check) all strut attach bolts on the forward and aft mounts. See Doc. D626A001-DTR, DTR check form 54-55-13, for alternative inspections.						
54-640-01-01	AWL	54-05-02-230-804	1.1	56000 FC	75000 FC	ALL	ALL
	Inspect (Penetrant) the strut attach bolts on the forward and aft mounts. See Doc. D626A001-DTR, DTR check form 54-55-13 for alternative repeat inspection. **ACCESS NOTE:** Removal of engine and engine mounts is required.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
54-640-01-02	AWL	54-05-02-230-804	1.1	56000 FC	75000 FC	ALL	ALL
	Inspect (Penetrant) the strut attach bolts on the forward and aft mounts. See Doc. D626A001-DTR, DTR check form 54-55-13 for alternative repeat inspection. **ACCESS NOTE:** Removal of engine and engine mounts is required.						
54-800-01-01	MRB	05-41-04-210-811	1.1 1.2	9000 FC 36 MO	9000 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the forward strut fairing - engine no. 1. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 18000 FC/6 YR is satisfied by this zonal inspection.						
54-802-01-01	MRB	05-41-04-210-812	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the fan cowl support beam - engine no. 1. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 18000 FC/6 YR is satisfied by this zonal inspection.						
54-804-01-01	MRB	05-41-04-210-813	1.1 1.2	18000 FC 6 YR	18000 FC 6 YR	ALL	ALL
	Perform an internal zonal inspection (GV) of the strut torque box - engine no. 1. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 18000 FC/6 YR is satisfied by this zonal inspection.						
54-806-01-01	MRB	05-41-04-210-814	1.1 1.2	9000 FC 36 MO	9000 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the aft strut fairing - engine no. 1. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 12000 FC/4 YR is satisfied by this zonal inspection.						
54-808-02-01	MRB	05-41-04-210-815	1.1 1.2	9000 FC 36 MO	9000 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the forward strut fairing - engine no. 2. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 18000 FC/6 YR is satisfied by this zonal inspection.						
54-810-02-01	MRB	05-41-04-210-816	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the fan cowl support beam - engine no. 2. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 18000 FC/6 YR is satisfied by this zonal inspection.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
54-812-02-01	MRB	05-41-04-210-817	1.1 1.2	18000 FC 6 YR	18000 FC 6 YR	ALL	ALL
	Perform an internal zonal inspection (GV) of the strut torque box - engine no. 2. (EZAP)						
	INTERVAL NOTE: Whichever comes first. The EZAP inspection requirement with interval 18000 FC/6 YR is satisfied by this zonal inspection.						
54-814-02-01	MRB	05-41-04-210-818	1.1 1.2	9000 FC 36 MO	9000 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the aft strut fairing - engine no. 2. (EZAP)						
	INTERVAL NOTE: Whichever comes first. The EZAP inspection requirement with interval 12000 FC/4 YR is satisfied by this zonal inspection.						
55-010-00-01	MRB	51-05-01-210-809 55-05-03-210-801	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL
	Inspect horizontal stabilizer center section jackscrew fitting.						
	INTERVAL NOTE: Whichever comes first.						
55-020-00-01	MRB	51-05-01-210-804 55-05-03-210-802	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL
	Inspect horizontal stabilizer center section front spar, rear spar, pivot fittings, and primary and secondary thrust beams.						
	INTERVAL NOTE: Whichever comes first.						
	ACCESS NOTE: Adjust stabilizer trim as required. Remove gap seal and rear spar sliding seal to inspect pivot fittings.						
55-030-00-01	MRB	51-05-01-210-806 55-05-03-210-803	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL
	Inspect the forward side of the vertical fin front spar, including front spar chords, webs, and terminal fittings.						
	INTERVAL NOTE: Whichever comes first.						
55-050-00-01	MRB	51-05-01-210-804 55-05-03-210-804	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL
	Inspect vertical fin from front spar to rear spar, including spar chords and webs, inspar skins, lower closure rib, and rear spar terminal fittings.						
	INTERVAL NOTE: Whichever comes first.						
	ACCESS NOTE: Remove upper removable web. Pin removal is not required for terminal fitting inspection.						
55-060-00-01	MRB	51-05-01-210-806 55-05-03-210-805	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL
	Inspect aft side of vertical fin rear spar, including rear spar chords and webs, and rudder hinge ribs.						
	INTERVAL NOTE: Whichever comes first.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-070-01-01	MRB	51-05-01-210-804 55-05-03-210-806	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL

Inspect forward side of front spar, including front spar chords and webs, terminal fittings, and center section front spar lugs.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Bolt removal is required to inspect terminal fittings and center section front spar lugs.

55-070-02-01	MRB	51-05-01-210-804 55-05-03-210-807	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL

Inspect forward side of front spar, including front spar chords and webs, terminal fittings, and center section front spar lugs.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Bolt removal is required to inspect terminal fittings and center section front spar lugs.

55-080-01-01	MRB	51-05-01-210-801 55-05-03-210-808	1.1	8 YR	4 YR	ALL	ALL

Inspect left horizontal stabilizer front spar terminal fittings and center section front spar lugs.

ACCESS NOTE: Bolt removal is not required.

55-080-02-01	MRB	51-05-01-210-801 55-05-03-210-809	1.1	8 YR	4 YR	ALL	ALL

Inspect right horizontal stabilizer front spar terminal fittings and center section front spar lugs.

ACCESS NOTE: Bolt removal is not required.

55-100-01-01	MRB	51-05-01-210-806 55-05-03-210-810	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL

Inspect left horizontal stabilizer from front spar to rear spar including spar chords, webs, terminal fittings, upper and lower inspar skins.

INTERVAL NOTE: Whichever comes first.

55-100-02-01	MRB	51-05-01-210-806 55-05-03-210-811	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL

Inspect right horizontal stabilizer from front spar to rear spar including spar chords, webs, terminal fittings, upper and lower inspar skins.

INTERVAL NOTE: Whichever comes first.

55-110-01-01	MRB	51-05-01-210-804 55-05-03-210-812	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL

Inspect aft side of left horizontal stabilizer rear spar, including spar chords and webs, terminal fittings, elevator tab actuator support fitting, center section rear spar lugs, elevator hinge ribs, and elevator tab leading edge.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Disconnect balance panels in balance bays. Bolt removal is required to inspect terminal fittings and center section rear spar lugs, except at rear spar upper lugs and clevis. Remove tab hinge covers to inspect elevator tab leading edge.

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-110-02-01	MRB	51-05-01-210-804 55-05-03-210-813	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL

Inspect aft side of right horizontal stabilizer rear spar, including spar chords and webs, terminal fittings, elevator tab actuator support fitting, center section rear spar lugs, elevator hinge ribs, and elevator tab leading edge.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Disconnect balance panels in balance bays. Bolt removal is required to inspect terminal fittings and center section rear spar lugs, except at rear spar upper lugs and clevis. Remove tab hinge covers to inspect elevator tab leading edge.

55-115-01-01	MRB	51-05-01-210-804 55-05-03-211-801	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL

Inspect left elevator tab mechanism support fittings on elevator front spar and elevator tab spar at leading edge cutouts.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Remove upper or lower horizontal stabilizer trailing edge seal and elevator inboard hinge cover panel to inspect support fittings. Remove tab hinge covers to locally inspect forward face of spar.

55-115-02-01	MRB	51-05-01-210-804 55-05-03-211-802	1.1 1.2	12 YR 36000 FC	8 YR 24000 FC	ALL	ALL

Inspect right elevator tab mechanism support fittings on elevator front spar and elevator tab spar at leading edge cutouts.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Remove upper or lower horizontal stabilizer trailing edge seal and elevator inboard hinge cover panel to inspect support fittings. Remove tab hinge covers to locally inspect forward face of spar.

55-120-01-01	MRB	51-05-01-210-801 55-05-03-210-814	1.1	8 YR	4 YR	ALL	ALL

Inspect left horizontal stabilizer rear spar terminal fittings and center section rear spar lugs.

ACCESS NOTE: Bolt removal is not required.

55-120-02-01	MRB	51-05-01-210-801 55-05-03-210-815	1.1	8 YR	4 YR	ALL	ALL

Inspect right horizontal stabilizer rear spar terminal fittings and center section rear spar lugs.

ACCESS NOTE: Bolt removal is not required.

55-130-00-01	MRB	51-05-01-210-806 55-05-03-210-816	1.1 1.2	12 YR 36000 FC	10 YR 30000 FC	ALL	ALL

Inspect rudder, elevator and elevator tab skin panels, rudder and elevator spars, rudder and elevator inspar ribs, rudder and elevator leading edge skins, rudder and elevator leading edge ribs, and rudder and elevator leading edge spars.

INTERVAL NOTE: Whichever comes first.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-135-01-01	MRB	51-05-01-210-803 55-05-03-211-807	1.1 1.2	8 YR 18000 FC	8 YR 18000 FC	ALL	ALL

Inspect the left elevator hinge fittings, left elevator actuator fittings, left elevator balance weight support structure, left elevator tab mast arm fitting and left elevator tab hinge fittings.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: After removal of access panels, disconnect balance bays. Alternative - Remove elevator from horizontal stabilizer. For elevator tab mast arm fittings, remove fairing on upper surface of elevator to gain access.

55-135-02-01	MRB	51-05-01-210-803 55-05-03-211-808	1.1 1.2	8 YR 18000 FC	8 YR 18000 FC	ALL	ALL

Inspect the right elevator hinge fittings, right elevator actuator fittings, right elevator balance weight support structure, right elevator tab mast arm fitting and right elevator tab hinge fittings.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: After removal of access panels, disconnect balance bays. Alternative - Remove elevator from horizontal stabilizer. For elevator tab mast arm fittings, remove fairing on upper surface of elevator to gain access.

55-600-00-01	AWL	55-05-02-130-801	1.1	56000 FC	4000 FC	ALL	ALL

Inspect (Ultrasonic) the spar chord between the web and shear tie at BL 1.3 on the left and right hand sides.
See Doc. D626A001 - DTR, DTR check form 55-10-01-1, alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 55-10-06.

ACCESS NOTE: Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

55-600-00-02	AWL	55-05-02-130-801	1.1	56000 FC	4000 FC	ALL	ALL

Inspect (Ultrasonic) the spar chord between the web and shear tie at BL 1.3 on the left and right hand sides.
See Doc. D626A001 - DTR, DTR check form 55-10-01-1, alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 55-10-06.

ACCESS NOTE: Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

55-601-00-01	AWL	55-05-02-250-801	1.1	56000 FC	4000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the upper chord around the fasteners common to the web at BL 2.7 and BL 6.7 on the left and right hand sides.
See Doc. D626A001 - DTR, DTR check form 55-10-01-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-16.

ACCESS NOTE: Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-601-00-02	AWL	55-05-02-250-801	1.1	56000 FC	4000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the upper chord around the fasteners common to the web at BL 2.7 and BL 6.7 on the left and right hand sides.
See Doc. D626A001 - DTR, DTR check form 55-10-01-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-16.

ACCESS NOTE: Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

55-602-00-01	AWL	55-05-02-250-802	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the rear spar upper chord from BL 8.1 to BL 19.7 on the left and right sides.
See Doc. D626A001 - DTR, DTR check form 55-10-01-3, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-22.

ACCESS NOTE: Removal of fasteners from both LBL and RBL 8.1 to 19.7 is required.
Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

55-602-00-02	AWL	55-05-02-250-802	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the rear spar upper chord from BL 8.1 to BL 19.7 on the left and right sides.
See Doc. D626A001 - DTR, DTR check form 55-10-01-3, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-22.

ACCESS NOTE: Removal of fasteners from both LBL and RBL 8.1 to 19.7 is required.
Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

55-603-00-01	AWL	55-05-02-130-802	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (Ultrasonic) the rear spar forward flange at the four Texas Star attachment points.
See Doc. D626A001 - DTR, DTR check form 55-10-01-4, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 55-10-07.

ACCESS NOTE: Access horizontal stabilizer center section through opening in center of 1088 bulkhead

55-603-00-02	AWL	55-05-02-130-802	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (Ultrasonic) the rear spar forward flange at the four Texas Star attachment points.
See Doc. D626A001 - DTR, DTR check form 55-10-01-4, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 55-10-07.

ACCESS NOTE: Access horizontal stabilizer center section through opening in center of 1088 bulkhead

55-604-00-01	AWL	55-05-02-130-804	1.1	56000 FC	10000 FC	ALL	ALL

Inspect (Ultrasonic) the pivot fittings on the horizontal stabilizer center section hinge house pivot lug.
See Doc. D626A001 - DTR, DTR check form 55-10-02-1, for alternative inspection.

ACCESS NOTE: Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-604-00-02	AWL	55-05-02-130-804	1.1	56000 FC	10000 FC	ALL	ALL

Inspect (Ultrasonic) the pivot fittings on the horizontal stabilizer center section hinge house pivot lug.
See Doc. D626A001 - DTR, DTR check form 55-10-02-1, for alternative inspection.

ACCESS NOTE: Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

55-605-00-01	AWL	55-05-02-130-824	1.1	56000 FC	3000 FC	ALL	ALL

Inspect (Ultrasonic) the pivot fitting plates at the lower hinge housing lugs.
See Doc. D626A001 - DTR, DTR check form 55-10-02-2, for alternative inspection.

ACCESS NOTE: Removal of gap covers and sliding seals is required.

55-605-00-02	AWL	55-05-02-130-824	1.1	56000 FC	3000 FC	ALL	ALL

Inspect (Ultrasonic) the pivot fitting plates at the lower hinge housing lugs.
See Doc. D626A001 - DTR, DTR check form 55-10-02-2, for alternative inspection.

ACCESS NOTE: Removal of gap covers and sliding seals is required.

55-606-00-01	AWL	55-05-02-130-819	1.1	56000 FC	1600 FC	ALL	ALL

Inspect (Ultrasonic) the fitting lugs on the horizontal stabilizer center section thrust link fittings. There are 4 lugs per fitting.
See Doc. D626A001 - DTR, DTR check form 55-10-02-3, for alternative inspection.

ACCESS NOTE: Remove fittings to inspect the fitting lugs, 4 lugs per fitting.
Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

55-606-00-02	AWL	55-05-02-130-819	1.1	56000 FC	1600 FC	ALL	ALL

Inspect (Ultrasonic) the fitting lugs on the horizontal stabilizer center section thrust link fittings. There are 4 lugs per fitting.
See Doc. D626A001 - DTR, DTR check form 55-10-02-3, for alternative inspection.

ACCESS NOTE: Remove fittings to inspect the fitting lugs, 4 lugs per fitting.
Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

55-607-00-01	AWL	55-05-02-130-823	1.1	56000 FC	6000 FC	ALL	ALL

Inspect (Ultrasonic) the horizontal stabilizer center section jackscrew fitting lugs.
See Doc. D626A001 - DTR, DTR check form 55-10-03-1, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 55-10-04.

ACCESS NOTE: Removal of bolts and bushings required.

55-607-00-02	AWL	55-05-02-130-823	1.1	56000 FC	6000 FC	ALL	ALL

Inspect (Ultrasonic) the horizontal stabilizer center section jackscrew fitting lugs.
See Doc. D626A001 - DTR, DTR check form 55-10-03-1, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 55-10-04.

ACCESS NOTE: Removal of bolts and bushings required.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-608-00-01	AWL	55-05-02-250-806	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the visible portion of the lower chord between stabilizer STA 129.5 and stabilizer STA 212.3.
See Doc. D626A001 - DTR, DTR check form 55-10-04-1, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-27.

55-608-00-02	AWL	55-05-02-250-806	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the visible portion of the lower chord between stabilizer STA 129.5 and stabilizer STA 212.3.
See Doc. D626A001 - DTR, DTR check form 55-10-04-1, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-27.

55-609-00-01	AWL	55-05-02-250-807	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) both fastener rows in the FWD flange at, and between, the ribs at stabilizer STA 212.3 to stabilizer STA 310.54.
See Doc. D626A001 - DTR, DTR check form 55-10-04-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-29.

55-609-00-02	AWL	55-05-02-250-807	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) both fastener rows in the FWD flange at, and between, the ribs at stabilizer STA 212.3 to stabilizer STA 310.54.
See Doc. D626A001 - DTR, DTR check form 55-10-04-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-29.

55-610-00-01	AWL	55-05-02-250-805	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the rear spar lower chord AFT flange, at and between the ribs, from stabilizer STA 212.3 to stabilizer STA 310.54.
See Doc. D626A001 - DTR, DTR check form 55-10-04-3 for alternative inspection.

ACCESS NOTE: Inspection requires removal of the Trailing Edge (TE) panel and fasteners.
The TE panels are attached by nutplates.

55-610-00-02	AWL	55-05-02-250-805	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the rear spar lower chord AFT flange, at and between the ribs, from stabilizer STA 212.3 to stabilizer STA 310.54.
See Doc. D626A001 - DTR, DTR check form 55-10-04-3 for alternative inspection.

ACCESS NOTE: Inspection requires removal of the Trailing Edge (TE) panel and fasteners.
The TE panels are attached by nutplates.

55-611-00-01	AWL	55-05-02-130-803	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (Ultrasonic) the rear spar lower chord web flange at the Trailing Edge (TE) ribs and stiffeners from stabilizer STA 212.3 to stabilizer STA 310.54.
See Doc. D626A001 - DTR, DTR check form 55-10-04-4, for alternative inspection.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-611-00-02	AWL	55-05-02-130-803	1.1	56000 FC	36000 FC	ALL	ALL
	Inspect (Ultrasonic) the rear spar lower chord web flange at the Trailing Edge (TE) ribs and stiffeners from stabilizer STA 212.3 to stabilizer STA 310.54. See Doc. D626A001 - DTR, DTR check form 55-10-04-4, for alternative inspection.						
55-611-10-01	AWL	55-05-02-250-845	1.1	56000 FC	24000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the rear spar lower chord web flange between the Trailing Edge (TE) ribs from stabilizer STA 212.3 to stabilizer STA 310.54. See Doc. D626A001 - DTR, DTR check form 55-10-04-5, for alternative inspection.						
55-611-10-02	AWL	55-05-02-250-845	1.1	56000 FC	24000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the rear spar lower chord web flange between the Trailing Edge (TE) ribs from stabilizer STA 212.3 to stabilizer STA 310.54. See Doc. D626A001 - DTR, DTR check form 55-10-04-5, for alternative inspection.						
55-611-12-01	AWL	55-05-02-250-846	1.1	56000 FC	24000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) both the fastener rows on the rear spar lower chord FWD flange, at and between the ribs, from stabilizer STA 310.54 to outboard tip. See Doc. D626A001 - DTR, DTR check form 55-10-04-6, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-56.						
55-611-12-02	AWL	55-05-02-250-846	1.1	56000 FC	24000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) both the fastener rows on the rear spar lower chord FWD flange, at and between the ribs, from stabilizer STA 310.54 to outboard tip. See Doc. D626A001 - DTR, DTR check form 55-10-04-6, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-56.						
55-611-14-01	AWL	55-05-02-250-815	1.1	56000 FC	15000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) the rear spar lower chord aft flange, at and between the ribs, from stabilizer STA 310.54 to outboard tip. See Doc. D626A001-DTR, DTR check form 55-10-04-7, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-55.						
55-611-14-02	AWL	55-05-02-250-815	1.1	56000 FC	15000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) the rear spar lower chord aft flange, at and between the ribs, from stabilizer STA 310.54 to outboard tip. See Doc. D626A001-DTR, DTR check form 55-10-04-7, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-55.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-611-16-01	AWL	55-05-02-250-847	1.1	56000 FC	24000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) the rear spar lower chord from stabilizer STA 310.54 to outboard tip web flange, at and between, the ribs. See Doc. D626A001 - DTR, DTR check form 55-10-04-8, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-50.						
55-611-16-02	AWL	55-05-02-250-847	1.1	56000 FC	24000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) the rear spar lower chord from stabilizer STA 310.54 to outboard tip web flange, at and between, the ribs. See Doc. D626A001 - DTR, DTR check form 55-10-04-8, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-50.						
55-611-18-01	AWL	55-05-02-250-848	1.1	56000 FC	36000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the rear spar lower chord at Bay 21- AFT flange at the hinge, from stabilizer STA 242 to stabilizer STA 247, and elevator STA 197 to elevator STA 203. See Doc. D626A001 - DTR, DTR check form 55-10-04-9, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-52.						
55-611-18-02	AWL	55-05-02-250-848	1.1	56000 FC	36000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the rear spar lower chord at Bay 21- AFT flange at the hinge, from stabilizer STA 242 to stabilizer STA 247, and elevator STA 197 to elevator STA 203. See Doc. D626A001 - DTR, DTR check form 55-10-04-9, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-52.						
55-611-20-01	AWL	55-05-02-250-849	1.1	56000 FC	2000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the rear spar upper chord between the terminal fitting fork. Note: Inspection is applicable to all four bolt locations common to the chord and terminal fitting. See Doc. D626A001 - DTR, DTR check form 55-10-04-10A, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-27.						
55-611-20-02	AWL	55-05-02-250-849	1.1	56000 FC	2000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the rear spar upper chord between the terminal fitting fork. Note: Inspection is applicable to all four bolt locations common to the chord and terminal fitting. See Doc. D626A001 - DTR, DTR check form 55-10-04-10A, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-27.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-611-22-01	AWL	55-05-02-250-804	1.1	56000 FC	3550 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the rear spar upper chord between the terminal fitting fork.
Note: Inspection is applicable to the outboard three bolt locations common to the chord and terminal fitting.
See Doc. D626A001-DTR, DTR check form 55-10-04-10B, for alternative inspection.

ACCESS NOTE: The aft side is accessed from the trailing edge. The forward side is accessed through the access panel in the lower inspar skin. Sealant removal is required.

55-611-22-02	AWL	55-05-02-250-804	1.1	56000 FC	3550 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the rear spar upper chord between the terminal fitting fork.
Note: Inspection is applicable to the outboard three bolt locations common to the chord and terminal fitting.
See Doc. D626A001-DTR, DTR check form 55-10-04-10B, for alternative inspection.

ACCESS NOTE: The aft side is accessed from the trailing edge. The forward side is accessed through the access panel in the lower inspar skin. Sealant removal is required.

55-611-23-01	AWL	55-05-02-130-814	1.1	56000 FC	3550 FC	ALL	ALL

Inspect (Ultrasonic) the rear spar upper chord between the terminal fitting fork.
Note: Inspection is applicable to the outboard three bolt locations common to the chord and terminal fitting.
See Doc. D626A001 - DTR, DTR check form 55-10-04-10B, for alternative inspection.

ACCESS NOTE: The aft side is accessed from the trailing edge. The forward side is accessed through the access panel in the lower inspar skin. Sealant removal is required.

55-611-23-02	AWL	55-05-02-130-814	1.1	56000 FC	3550 FC	ALL	ALL

Inspect (Ultrasonic) the rear spar upper chord between the terminal fitting fork.
Note: Inspection is applicable to the outboard three bolt locations common to the chord and terminal fitting.
See Doc. D626A001 - DTR, DTR check form 55-10-04-10B, for alternative inspection.

ACCESS NOTE: The aft side is accessed from the trailing edge. The forward side is accessed through the access panel in the lower inspar skin. Sealant removal is required.

55-611-24-01	AWL	55-05-02-250-850	1.1	56000 FC	12000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the rear spar upper chord between the terminal fitting fork.
Note: Inspection is applicable to the inboard bolt location common to the chord and terminal fitting.
See Doc. D626A001 - DTR, DTR check form 55-10-04-10C, for alternative inspection.

55-611-24-02	AWL	55-05-02-250-850	1.1	56000 FC	12000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the rear spar upper chord between the terminal fitting fork.
Note: Inspection is applicable to the inboard bolt location common to the chord and terminal fitting.
See Doc. D626A001 - DTR, DTR check form 55-10-04-10C, for alternative inspection.

55-611-25-01	AWL	55-05-02-130-816	1.1	56000 FC	12000 FC	ALL	ALL

Inspect (Ultrasonic) the rear spar upper chord between the terminal fitting fork.
Note: Inspection is applicable to the inboard bolt location common to the chord and terminal fitting.
See Doc. D626A001 - DTR, DTR check form 55-10-04-10C, for alternative inspection.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-611-25-02	AWL	55-05-02-130-816	1.1	56000 FC	12000 FC	ALL	ALL
	Inspect (Ultrasonic) the rear spar upper chord between the terminal fitting fork. Note: Inspection is applicable to the inboard bolt location common to the chord and terminal fitting. See Doc. D626A001 - DTR, DTR check form 55-10-04-10C, for alternative inspection.						
55-611-26-01	AWL	55-05-02-250-851	1.1	56000 FC	2000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) both the forward and aft rear spar upper chord flanges from stabilizer STA 67.78 to stabilizer STA 203.10. See Doc. D626A001 - DTR, DTR check form 55-10-04-11A, for alternative inspection.						
55-611-26-02	AWL	55-05-02-250-851	1.1	56000 FC	2000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) both the forward and aft rear spar upper chord flanges from stabilizer STA 67.78 to stabilizer STA 203.10. See Doc. D626A001 - DTR, DTR check form 55-10-04-11A, for alternative inspection.						
55-611-27-01	AWL	55-05-02-250-852	1.1	56000 FC	3500 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the rear spar upper cord from stabilizer STA 67.78 to stabilizer STA 203.10. See Doc. D626A001 - DTR, DTR check form 55-10-04-11A, for alternative inspection.						
55-611-27-02	AWL	55-05-02-250-852	1.1	56000 FC	3500 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the rear spar upper cord from stabilizer STA 67.78 to stabilizer STA 203.10. See Doc. D626A001 - DTR, DTR check form 55-10-04-11A, for alternative inspection.						
55-611-28-01	AWL	55-05-02-250-853	1.1	56000 FC	15000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) both the forward and aft rear spar upper chord skin flanges from stabilizer STA 203.10 to stabilizer STA 258.28. See Doc. D626A001 - DTR, DTR check form 55-10-04-11B, for alternative inspection.						
55-611-28-02	AWL	55-05-02-250-853	1.1	56000 FC	15000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) both the forward and aft rear spar upper chord skin flanges from stabilizer STA 203.10 to stabilizer STA 258.28. See Doc. D626A001 - DTR, DTR check form 55-10-04-11B, for alternative inspection.						
55-611-30-01	AWL	55-05-02-130-817	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (Ultrasonic) the rear spar upper chord web flange at the rib and stiffener locations from stabilizer STA 203.1 to stabilizer STA 258.28. See Doc. D626A001 - DTR, DTR check form 55-10-04-11C, for alternative inspection.						
55-611-30-02	AWL	55-05-02-130-817	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (Ultrasonic) the rear spar upper chord web flange at the rib and stiffener locations from stabilizer STA 203.1 to stabilizer STA 258.28. See Doc. D626A001 - DTR, DTR check form 55-10-04-11C, for alternative inspection.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-611-32-01	AWL	55-05-02-250-854	1.1	56000 FC	24000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the rear spar upper chord web flange between the ribs and stiffeners from stabilizer STA 203.10 to stabilizer STA 258.28. Note: Remove sealant in excess of .20" on either side of the fastener head or collar. See Doc. D626A001 - DTR, DTR check form 55-10-04-11D, for alternative inspection.						
55-611-32-02	AWL	55-05-02-250-854	1.1	56000 FC	24000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the rear spar upper chord web flange between the ribs and stiffeners from stabilizer STA 203.10 to stabilizer STA 258.28. Note: Remove sealant in excess of .20" on either side of the fastener head or collar. See Doc. D626A001 - DTR, DTR check form 55-10-04-11D, for alternative inspection.						
55-611-34-01	AWL	55-05-02-250-855	1.1	56000 FC	15000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) the rear spar upper chord, at the forward flange of the chord, from stabilizer STA 258.28 to the tip. See Doc. D626A001 - DTR, DTR check form 55-10-04-12A, for alternative inspection.						
55-611-34-02	AWL	55-05-02-250-855	1.1	56000 FC	15000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) the rear spar upper chord, at the forward flange of the chord, from stabilizer STA 258.28 to the tip. See Doc. D626A001 - DTR, DTR check form 55-10-04-12A, for alternative inspection.						
55-611-36-01	AWL	55-05-02-250-856	1.1	56000 FC	21000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) the rear spar upper cord, at the aft flange of the cord, from stabilizer STA 258.28 to the tip. See Doc. D626A001 - DTR, DTR check form 55-10-04-12B, for alternative inspection.						
55-611-36-02	AWL	55-05-02-250-856	1.1	56000 FC	21000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) the rear spar upper cord, at the aft flange of the cord, from stabilizer STA 258.28 to the tip. See Doc. D626A001 - DTR, DTR check form 55-10-04-12B, for alternative inspection.						
55-611-38-01	AWL	55-05-02-130-818	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (Ultrasonic) the rear spar upper cord web flange, at the rib and stiffener locations, from stabilizer STA 258.28 to stabilizer STA 310.54. See Doc. D626A001 - DTR, DTR check form 55-10-04-13A, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Section 55-10-19.						
55-611-38-02	AWL	55-05-02-130-818	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (Ultrasonic) the rear spar upper cord web flange, at the rib and stiffener locations, from stabilizer STA 258.28 to stabilizer STA 310.54. See Doc. D626A001 - DTR, DTR check form 55-10-04-13A, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Section 55-10-19.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-611-40-01	AWL	55-05-02-250-871	1.1	56000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the rear spar upper chord web flange, between the rib and stiffener locations, from stabilizer STA 258.28 to stabilizer STA 310.54.
See Doc. D626A001 - DTR, DTR check form 55-10-04-13B, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-59.

55-611-40-02	AWL	55-05-02-250-871	1.1	56000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the rear spar upper chord web flange, between the rib and stiffener locations, from stabilizer STA 258.28 to stabilizer STA 310.54.
See Doc. D626A001 - DTR, DTR check form 55-10-04-13B, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-59.

55-611-42-01	AWL	55-05-02-250-857	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the rear spar upper cord web flange, at and between the ribs/stiffeners, from stabilizer STA 310.54 to tip.
See Doc. D626A001 - DTR, DTR check form 55-10-04-13C, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-60.

ACCESS NOTE: Access is through the lower inspar skin access hole. Subsurface inspection of the chord is through the web or through the web and stiffener.

55-611-42-02	AWL	55-05-02-250-857	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the rear spar upper cord web flange, at and between the ribs/stiffeners, from stabilizer STA 310.54 to tip.
See Doc. D626A001 - DTR, DTR check form 55-10-04-13C, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-60.

ACCESS NOTE: Access is through the lower inspar skin access hole. Subsurface inspection of the chord is through the web or through the web and stiffener.

55-611-44-01	AWL	55-05-02-250-858	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the rear spar web at the lower edge of the fail-safe chord, at and between the stiffeners, from stabilizer STA 83.5 to stabilizer STA 184.7.
See Doc. D626A001 - DTR, DTR check form 55-10-04-20, for alternative inspection.

ACCESS NOTE: Access is through the removable trailing edge lower skin panels. The DTR curve assumes no fillet seal. Remove any fillet seals present.

55-611-44-02	AWL	55-05-02-250-858	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the rear spar web at the lower edge of the fail-safe chord, at and between the stiffeners, from stabilizer STA 83.5 to stabilizer STA 184.7.
See Doc. D626A001 - DTR, DTR check form 55-10-04-20, for alternative inspection.

ACCESS NOTE: Access is through the removable trailing edge lower skin panels. The DTR curve assumes no fillet seal. Remove any fillet seals present.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-611-46-01	AWL	55-05-02-250-859	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the rear spar web at the lower edge of the upper chord, at and between the stiffeners, from stabilizer STA 184.7 to stabilizer STA 285.9.

See Doc. D626A001 - DTR, DTR check form 55-10-04-21, for alternative inspection.

ACCESS NOTE: Access is through the removable Trailing Edge Lower Skin Panels. The DTR curve assumes no fillet seal. Remove any fillet seals present.

55-611-46-02	AWL	55-05-02-250-859	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the rear spar web at the lower edge of the upper chord, at and between the stiffeners, from stabilizer STA 184.7 to stabilizer STA 285.9.

See Doc. D626A001 - DTR, DTR check form 55-10-04-21, for alternative inspection.

ACCESS NOTE: Access is through the removable Trailing Edge Lower Skin Panels. The DTR curve assumes no fillet seal. Remove any fillet seals present.

55-611-48-01	AWL	55-05-02-250-860	1.1	56000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the rear spar web at the upper and lower chord edges, at and between the stiffeners, from stabilizer STA 285.9 to stabilizer STA 310.5.

See Doc. D626A001 - DTR, DTR check form 55-10-04-22, for alternative inspection.

ACCESS NOTE: Access is through the removable Trailing Edge Panel Lower Skin Panels. The DTR curve assumes no fillet seal. Remove any fillet seals present.

55-611-48-02	AWL	55-05-02-250-860	1.1	56000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the rear spar web at the upper and lower chord edges, at and between the stiffeners, from stabilizer STA 285.9 to stabilizer STA 310.5.

See Doc. D626A001 - DTR, DTR check form 55-10-04-22, for alternative inspection.

ACCESS NOTE: Access is through the removable Trailing Edge Panel Lower Skin Panels. The DTR curve assumes no fillet seal. Remove any fillet seals present.

55-611-50-01	AWL	55-05-02-250-861	1.1	56000 FC	15000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the rear spar web at the upper and lower web to chord fastener locations, between the stiffeners, from stabilizer STA 310.5 to outboard tip.

See Doc. D626A001 - DTR, DTR check form 55-10-04-23, for alternative inspection.

ACCESS NOTE: Access is through the inspar lower skin access panels. The DTR curve assumes no cap seal. Remove any cap seals present.

55-611-50-02	AWL	55-05-02-250-861	1.1	56000 FC	15000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the rear spar web at the upper and lower web to chord fastener locations, between the stiffeners, from stabilizer STA 310.5 to outboard tip.

See Doc. D626A001 - DTR, DTR check form 55-10-04-23, for alternative inspection.

ACCESS NOTE: Access is through the inspar lower skin access panels. The DTR curve assumes no fillet seal. Remove any fillet seals present.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-611-52-01	AWL	55-05-02-130-820	1.1	56000 FC	15000 FC	ALL	ALL

Inspect (Ultrasonic) the rear spar web at the stiffeners common to both the upper and lower chords from stabilizer STA 310.5 to outboard tip.
See Doc. D626A001 - DTR, DTR check form 55-10-04-24, for alternative inspection.

ACCESS NOTE: Access is through the inspar lower skin access panels.

55-611-52-02	AWL	55-05-02-130-820	1.1	56000 FC	15000 FC	ALL	ALL

Inspect (Ultrasonic) the rear spar web at the stiffeners common to both the upper and lower chords from stabilizer STA 310.5 to outboard tip.
See Doc. D626A001 - DTR, DTR check form 55-10-04-24, for alternative inspection.

ACCESS NOTE: Access is through the inspar lower skin access panels.

55-612-00-01	AWL	55-05-02-130-821	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (Ultrasonic) the rear spar upper and failsafe chord lugs.
See Doc. D626A001 - DTR, DTR check form 55-10-05-1, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 55-10-15.

ACCESS NOTE: Removal of gap covers is required.

55-612-00-02	AWL	55-05-02-130-821	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (Ultrasonic) the rear spar upper and failsafe chord lugs.
See Doc. D626A001 - DTR, DTR check form 55-10-05-1, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 55-10-15.

ACCESS NOTE: Removal of gap covers is required.

55-613-00-01	AWL	55-05-02-130-822	1.1	56000 FC	3550 FC	ALL	ALL

Inspect (Ultrasonic) the forward and aft sides of the terminal fitting around the three inboard fasteners common to the spar chord.
Note: Bolts and bushings should remain installed for the inspection. Remove any sealant present.
See Doc. D626A001 - DTR, DTR check form 55-10-05-2A, for alternative inspection.

ACCESS NOTE: The aft side is accessed from the trailing edge. The forward side is accessed through the access panel in the inspar lower skin. Sealant, if present, must be removed for inspection.

55-613-00-02	AWL	55-05-02-130-822	1.1	56000 FC	3550 FC	ALL	ALL

Inspect (Ultrasonic) the forward and aft sides of the terminal fitting around the three inboard fasteners common to the spar chord.
Note: Bolts and bushings should remain installed for the inspection. Remove any sealant present.
See Doc. D626A001 - DTR, DTR check form 55-10-05-2A, for alternative inspection.

ACCESS NOTE: The aft side is accessed from the trailing edge. The forward side is accessed through the access panel in the inspar lower skin. Sealant, if present, must be removed for inspection.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-613-01-01	AWL	55-05-02-250-812	1.1	56000 FC	3550 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the forward and aft sides of the terminal fitting around the three inboard fasteners common to the spar chord.
Note: Bolts and bushings should remain installed for the inspection. Remove any sealant present.
See Doc. D626A001 - DTR, DTR check form 55-10-05-2A, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-18.

ACCESS NOTE: The aft side is accessed from the trailing edge. The forward side is accessed through the access panel in the inspar lower skin. Sealant, if present, must be removed for inspection.

55-613-01-02	AWL	55-05-02-250-812	1.1	56000 FC	3550 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the forward and aft sides of the terminal fitting around the three inboard fasteners common to the spar chord.
Note: Bolts and bushings should remain installed for the inspection. Remove any sealant present.
See Doc. D626A001 - DTR, DTR check form 55-10-05-2A, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-18.

ACCESS NOTE: The aft side is accessed from the trailing edge. The forward side is accessed through the access panel in the inspar lower skin. Sealant, if present, must be removed for inspection.

55-613-10-01	AWL	55-05-02-211-813	1.1	56000 FC	9000 FC	ALL	ALL

Inspect (Detailed) the forward and aft sides of the terminal fitting at the outboard attach bolt location.
Note: Bolts and bushings should remain installed for the inspection. Sealant removal is required.
See Doc. D626A001 - DTR, DTR check form 55-10-05-2B, for alternative inspection.

ACCESS NOTE: The aft side is accessed from the trailing edge. The forward side is accessed through the access panel in the lower inspar skin. Use of a Bore scope is required.

55-613-10-02	AWL	55-05-02-211-813	1.1	56000 FC	9000 FC	ALL	ALL

Inspect (Detailed) the forward and aft sides of the terminal fitting at the outboard attach bolt location.
Note: Bolts and bushings should remain installed for the inspection. Sealant removal is required.
See Doc. D626A001 - DTR, DTR check form 55-10-05-2B, for alternative inspection.

ACCESS NOTE: The aft side is accessed from the trailing edge. The forward side is accessed through the access panel in the lower inspar skin. Use of a Bore scope is required.

55-614-00-01	AWL	55-05-02-230-801	1.1	56000 FC	9000 FC	ALL	ALL

Inspect (Penetrant) the upper, lower and failsafe pins at the side of body on the horizontal stabilizer rear spar terminal fitting for both the left and right sides.
See Doc. D626A001 - DTR, DTR check form 55-10-05-3, for alternative inspection.

ACCESS NOTE: Removal of pins is required.

55-614-00-02	AWL	55-05-02-230-801	1.1	56000 FC	9000 FC	ALL	ALL

Inspect (Penetrant) the upper, lower and failsafe pins at the side of body on the horizontal stabilizer rear spar terminal fitting for both the left and right sides.
See Doc. D626A001 - DTR, DTR check form 55-10-05-3, for alternative inspection.

ACCESS NOTE: Removal of pins is required.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-614-10-01	AWL	55-05-02-250-810	1.1	56000 FC	21000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the rear spar upper chord terminal fitting on the upper side of the Side of Body (SOB) rib tension bolt hole.
See Doc. D626A001 - DTR, DTR check form 55-10-05-4, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-49.

ACCESS NOTE: Removal of the tension bolt is required.

55-614-10-02	AWL	55-05-02-250-810	1.1	56000 FC	21000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the rear spar upper chord terminal fitting on the upper side of the Side of Body (SOB) rib tension bolt hole.
See Doc. D626A001 - DTR, DTR check form 55-10-05-4, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-49.

ACCESS NOTE: Removal of the tension bolt is required.

55-614-11-01	AWL	55-05-02-250-883	1.1	56000 FC	21000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the rear spar upper chord terminal fitting on the lower side of the Side of Body (SOB) rib tension bolt hole.
See Doc. D626A001 - DTR, DTR check form 55-10-05-4, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-49.

ACCESS NOTE: Removal of the tension bolt is required.

55-614-11-02	AWL	55-05-02-250-883	1.1	56000 FC	21000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the rear spar upper chord terminal fitting on the lower side of the Side of Body (SOB) rib tension bolt hole.
See Doc. D626A001 - DTR, DTR check form 55-10-05-4, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-49.

ACCESS NOTE: Removal of the tension bolt is required.

55-614-12-01	AWL	55-05-02-250-863	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the rear spar lower chord at the side of body, common to the Side of Body (SOB) rib tension bolt hole.
See Doc. D626A001 - DTR, DTR check form 55-10-05-5, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-48.

ACCESS NOTE: The inspection requires the removal of the bolt.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-614-12-02	AWL	55-05-02-250-863	1.1	56000 FC	36000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the rear spar lower chord at the side of body, common to the Side of Body (SOB) rib tension bolt hole. See Doc. D626A001 - DTR, DTR check form 55-10-05-5, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-48. **ACCESS NOTE:** The inspection requires the removal of the bolt.						
55-615-00-01	AWL	55-05-02-250-808	1.1	56000 FC	2000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) the front spar upper chord from the side of body to stabilizer STA 66.5. See Doc. D626A001 - DTR, DTR check form 55-10-06-1, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-28.						
55-615-00-02	AWL	55-05-02-250-808	1.1	56000 FC	2000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) the front spar upper chord from the side of body to stabilizer STA 66.5. See Doc. D626A001 - DTR, DTR check form 55-10-06-1, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-28.						
55-616-00-01	AWL	55-05-02-250-803	1.1	56000 FC	2000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) the front spar upper chord aft flange from stabilizer STA 66.5 to stabilizer tip. See Doc. D626A001 - DTR, DTR check form 55-10-06-2, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-32.						
55-616-00-02	AWL	55-05-02-250-803	1.1	56000 FC	2000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) the front spar upper chord aft flange from stabilizer STA 66.5 to stabilizer tip. See Doc. D626A001 - DTR, DTR check form 55-10-06-2, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-32.						
55-616-05-01	AWL	55-05-02-250-885	1.1	56000 FC	6000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) the front spar upper chord web flange from stabilizer STA 75.0 to Stab Sta 175.5, except at web splice, stabilizer STA 111.1. See Doc. D626A001 - DTR, DTR check form 55-10-06-4C, for alternative inspection. **ACCESS NOTE:** Removal of the Leading Edge is required for inspection access.						
55-616-05-02	AWL	55-05-02-250-885	1.1	56000 FC	6000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) the front spar upper chord web flange from stabilizer STA 75.0 to Stab Sta 175.5, except at web splice, stabilizer STA 111.1. See Doc. D626A001 - DTR, DTR check form 55-10-06-4C, for alternative inspection. **ACCESS NOTE:** Removal of the Leading Edge is required for inspection access.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-616-06-01	AWL	55-05-02-250-886	1.1	56000 FC	6000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the front spar upper chord web flange from stabilizer STA 75.0 to Stab Sta 175.5, except at web splice, stabilizer STA 111.1. See Doc. D626A001 - DTR, DTR check form 55-10-06-4C, for alternative inspection. **ACCESS NOTE:** Removal of the Leading Edge is required for inspection access.						
55-616-06-02	AWL	55-05-02-250-886	1.1	56000 FC	6000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the front spar upper chord web flange from stabilizer STA 75.0 to Stab Sta 175.5, except at web splice, stabilizer STA 111.1. See Doc. D626A001 - DTR, DTR check form 55-10-06-4C, for alternative inspection. **ACCESS NOTE:** Removal of the Leading Edge is required for inspection access.						
55-616-07-01	AWL	55-05-02-250-887	1.1	56000 FC	15000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) the front spar upper chord web flange at web splice from STA 110.24 to STA 111.96. See Doc. D626A001 - DTR, DTR check form 55-10-06-4D, for alternative inspection. **ACCESS NOTE:** Removal of the Leading Edge is required for inspection access.						
55-616-07-02	AWL	55-05-02-250-887	1.1	56000 FC	15000 FC	ALL	ALL
	Inspect (Low Frequency Eddy Current) the front spar upper chord web flange at web splice from STA 110.24 to STA 111.96. See Doc. D626A001 - DTR, DTR check form 55-10-06-4D, for alternative inspection. **ACCESS NOTE:** Removal of the Leading Edge is required for inspection access.						
55-616-08-01	AWL	55-05-02-250-888	1.1	56000 FC	15000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the front spar upper chord web flange at web splice from STA 110.24 to STA 111.96. See Doc. D626A001 - DTR, DTR check form 55-10-06-4D, for alternative inspection. **ACCESS NOTE:** Removal of the Leading Edge is required for inspection access.						
55-616-08-02	AWL	55-05-02-250-888	1.1	56000 FC	15000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the front spar upper chord web flange at web splice from STA 110.24 to STA 111.96. See Doc. D626A001 - DTR, DTR check form 55-10-06-4D, for alternative inspection. **ACCESS NOTE:** Removal of the Leading Edge is required for inspection access.						
55-616-09-01	AWL	55-05-02-250-884	1.1	56000 FC	6000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the front spar upper chord web flange from stabilizer STA 258.28 to Outboard Tip. See Doc. D626A001 - DTR, DTR check form 55-10-06-4E, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-54. **ACCESS NOTE:** Removal of the Leading Edge is required for inspection access.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-616-09-02	AWL	55-05-02-250-884	1.1	56000 FC	6000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the front spar upper chord web flange from stabilizer STA 258.28 to Outboard Tip.
See Doc. D626A001 - DTR, DTR check form 55-10-06-4E, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-54.

ACCESS NOTE: Removal of the Leading Edge is required for inspection access.

55-616-10-01	AWL	55-05-02-250-864	1.1	56000 FC	6000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the front spar upper chord forward flange from stabilizer STA 66.5 to stabilizer tip.
Note: Inspection requires probe placement on primed metal surface.
See Doc. D626A001 - DTR, DTR check form 55-10-06-3, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-46.

ACCESS NOTE: Removal of the horizontal stabilizer removable leading edge is required.

55-616-10-02	AWL	55-05-02-250-864	1.1	56000 FC	6000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the front spar upper chord forward flange from stabilizer STA 66.5 to stabilizer tip.
Note: Inspection requires probe placement on primed metal surface.
See Doc. D626A001 - DTR, DTR check form 55-10-06-3, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-46.

ACCESS NOTE: Removal of the horizontal stabilizer removable leading edge is required.

55-616-11-01	AWL	55-05-02-250-889	1.1	56000 FC	2000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the front spar upper chord forward flange from stabilizer STA 66.5 to stabilizer tip.
Note: Inspection requires probe placement on primed metal surface.
See Doc. D626A001 - DTR, DTR check form 55-10-06-3, for alternative inspection.

ACCESS NOTE: Removal of the horizontal stabilizer removable leading edge is required.

55-616-11-02	AWL	55-05-02-250-889	1.1	56000 FC	2000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the front spar upper chord forward flange from stabilizer STA 66.5 to stabilizer tip.
Note: Inspection requires probe placement on primed metal surface.
See Doc. D626A001 - DTR, DTR check form 55-10-06-3, for alternative inspection.

ACCESS NOTE: Removal of the horizontal stabilizer removable leading edge is required.

55-616-12-01	AWL	55-05-02-250-865	1.1	56000 FC	6000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the front spar upper chord web flange from stabilizer STA 66.5 to stabilizer STA 75.0.
See Doc. D626A001 - DTR, DTR check form 55-10-06-4A, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-45.

ACCESS NOTE: Removal of the Leading Edge is required for inspection access.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-616-12-02	AWL	55-05-02-250-865	1.1	56000 FC	6000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the front spar upper chord web flange from stabilizer STA 66.5 to stabilizer STA 75.0.
See Doc. D626A001 - DTR, DTR check form 55-10-06-4A, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-45.

ACCESS NOTE: Removal of the Leading Edge is required for inspection access.

55-616-14-01	AWL	55-05-02-250-868	1.1	56000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the front spar upper chord web flange from stabilizer STA 175.5 to stabilizer STA 258.28.
Note: Inspection requires probe placement on primed metal surface.
See Doc. D626A001 - DTR, DTR check form 55-10-06-4B, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-47.

ACCESS NOTE: Removal of the horizontal stabilizer leading edge and the removable front spar web is required for inspection access.

55-616-14-02	AWL	55-05-02-250-868	1.1	56000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the front spar upper chord web flange from stabilizer STA 175.5 to stabilizer STA 258.28.
Note: Inspection requires probe placement on primed metal surface.
See Doc. D626A001 - DTR, DTR check form 55-10-06-4B, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-47.

ACCESS NOTE: Removal of the horizontal stabilizer leading edge and the removable front spar web is required for inspection access.

55-616-16-01	AWL	55-05-02-211-804	1.1	56000 FC	24000 FC	ALL	ALL

Inspect (Detailed) the front spar lower chord forward flange from stabilizer STA 66.50 to stabilizer STA 175.50.
See Doc. D626A001 - DTR, DTR check form 55-10-06-6A, for alternative inspection.

ACCESS NOTE: Removal of the leading edge panels is required to gain access to the lower surface of the forward flange.

55-616-16-02	AWL	55-05-02-211-804	1.1	56000 FC	24000 FC	ALL	ALL

Inspect (Detailed) the front spar lower chord forward flange from stabilizer STA 66.50 to stabilizer STA 175.50.
See Doc. D626A001 - DTR, DTR check form 55-10-06-6A, for alternative inspection.

ACCESS NOTE: Removal of the leading edge panels is required to gain access to the lower surface of the forward flange.

55-616-18-01	AWL	55-05-02-250-869	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the front spar lower chord aft flange from stabilizer STA 66.50 to stabilizer STA 175.50.
See Doc. D626A001 - DTR, DTR check form 55-10-06-6B, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-53.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-616-18-02	AWL	55-05-02-250-869	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the front spar lower chord aft flange from stabilizer STA 66.50 to stabilizer STA 175.50.

See Doc. D626A001 - DTR, DTR check form 55-10-06-6B, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-53.

55-616-20-01	AWL	55-05-02-250-870	1.1	56000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the front spar lower chord forward flange from stabilizer STA 175.50 to stabilizer STA 258.28.

See Doc. D626A001 - DTR, DTR check form 55-10-06-7A, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-51.

ACCESS NOTE: Removal of the leading edge panel is required to gain access to the lower surface of the forward flange.

55-616-20-02	AWL	55-05-02-250-870	1.1	56000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the front spar lower chord forward flange from stabilizer STA 175.50 to stabilizer STA 258.28.

See Doc. D626A001 - DTR, DTR check form 55-10-06-7A, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-51.

ACCESS NOTE: Removal of the leading edge panel is required to gain access to the lower surface of the forward flange.

55-616-22-01	AWL	55-05-02-250-872	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the front spar lower chord aft flange from stabilizer STA 175.50 to stabilizer STA 258.28.

See Doc. D626A001 - DTR, DTR check form 55-10-06-7B, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-57.

55-616-22-02	AWL	55-05-02-250-872	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the front spar lower chord aft flange from stabilizer STA 175.50 to stabilizer STA 258.28.

See Doc. D626A001 - DTR, DTR check form 55-10-06-7B, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-57.

55-616-24-01	AWL	55-05-02-250-874	1.1	56000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the front spar lower chord forward flange from stabilizer STA 258.28 to stabilizer BL 281.81.

See Doc. D626A001 - DTR, DTR check form 55-10-06-8A, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-51.

ACCESS NOTE: Leading edge skin assembly removal is required for the inspection.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-616-24-02	AWL	55-05-02-250-874	1.1	56000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the front spar lower chord forward flange from stabilizer STA 258.28 to stabilizer BL 281.81.
See Doc. D626A001 - DTR, DTR check form 55-10-06-8A, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-51.

ACCESS NOTE: Leading edge skin assembly removal is required for the inspection.

55-616-26-01	AWL	55-05-02-250-873	1.1	56000 FC	4000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the front spar lower chord aft flange from stabilizer STA 258.28 to stabilizer BL 281.81 (tip).
See Doc. D626A001 - DTR, DTR check form 55-10-06-8B, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-57.

55-616-26-02	AWL	55-05-02-250-873	1.1	56000 FC	4000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the front spar lower chord aft flange from stabilizer STA 258.28 to stabilizer BL 281.81 (tip).
See Doc. D626A001 - DTR, DTR check form 55-10-06-8B, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-57.

55-616-28-01	AWL	55-05-02-250-875	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the front spar web, at the upper and lower spar chord attachments, from stabilizer STA 66.5 to stabilizer STA 111.1.
See Doc. D626A001 - DTR, DTR check form 55-10-06-W1, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-58.

ACCESS NOTE: The inspection requires removal of the horizontal stabilizer leading edge.

55-616-28-02	AWL	55-05-02-250-875	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the front spar web, at the upper and lower spar chord attachments, from stabilizer STA 66.5 to stabilizer STA 111.1.
See Doc. D626A001 - DTR, DTR check form 55-10-06-W1, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-58.

ACCESS NOTE: The inspection requires removal of the horizontal stabilizer leading edge.

55-616-30-01	AWL	55-05-02-250-876	1.1	56000 FC	15000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the front spar web splice, at the upper and lower web edges at stabilizer STA 111.1.
Note: Inspection requires probe placement on primed metal surface.
See Doc. D626A001 - DTR, DTR check form 55-10-06-WS, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-58.

ACCESS NOTE: The inspection requires removal of the horizontal stabilizer leading edge.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-616-30-02	AWL	55-05-02-250-876	1.1	56000 FC	15000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the front spar web splice, at the upper and lower web edges at stabilizer STA 111.1.
Note: Inspection requires probe placement on primed metal surface.
See Doc. D626A001 - DTR, DTR check form 55-10-06-WS, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-58.

ACCESS NOTE: The inspection requires removal of the horizontal stabilizer leading edge.

55-616-32-01	AWL	55-05-02-211-814	1.1	56000 FC	12000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the front spar web, at the upper and lower spar chord attachments, from stabilizer STA 111.1 to stabilizer STA 175.5.
See Doc. D626A001 - DTR, DTR check form 55-10-06-W2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-58.

ACCESS NOTE: The inspection requires removal of the horizontal stabilizer leading edge. Remove any sealant which exceeds .30" on either side of a fastener head or collar.

55-616-32-02	AWL	55-05-02-211-814	1.1	56000 FC	12000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the front spar web, at the upper and lower spar chord attachments, from stabilizer STA 111.1 to stabilizer STA 175.5.
See Doc. D626A001 - DTR, DTR check form 55-10-06-W2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-58.

ACCESS NOTE: The inspection requires removal of the horizontal stabilizer leading edge. Remove any sealant which exceeds .30" on either side of a fastener head or collar.

55-616-34-01	AWL	55-05-02-250-878	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the front spar web, at the upper and lower spar chord attachments, from stabilizer STA 175.5 to stabilizer STA 258.28.
See Doc. D626A001 - DTR, DTR check form 55-10-06-W3, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-58.

ACCESS NOTE: The inspection requires removal of the horizontal stabilizer leading edge.
Remove any cap seal, that is present, on nut plated BACB30NM fasteners.

55-616-34-02	AWL	55-05-02-250-878	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the front spar web, at the upper and lower spar chord attachments, from stabilizer STA 175.5 to stabilizer STA 258.28.
See Doc. D626A001 - DTR, DTR check form 55-10-06-W3, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-58.

ACCESS NOTE: The inspection requires removal of the horizontal stabilizer leading edge.
Remove any cap seal, that is present, on nut plated BACB30NM fasteners.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-616-36-01	AWL	55-05-02-250-879	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the front spar web, at the upper and lower spar chord attachments, from stabilizer STA 258.28 to stabilizer tip.
See Doc. D626A001 - DTR, DTR check form 55-10-06-W4, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-58.

ACCESS NOTE: The inspection requires removal of the horizontal stabilizer leading edge.
Remove any cap sealant that is present.

55-616-36-02	AWL	55-05-02-250-879	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the front spar web, at the upper and lower spar chord attachments, from stabilizer STA 258.28 to stabilizer tip.
See Doc. D626A001 - DTR, DTR check form 55-10-06-W4, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-10-58.

ACCESS NOTE: The inspection requires removal of the horizontal stabilizer leading edge.
Remove any cap sealant that is present.

55-617-00-01	AWL	55-05-02-211-803	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (Detailed) the exposed surface of the front spar upper chord between the upper skin and Leading Edge (LE) skin.
See Doc. D626A001 - DTR, DTR check form 55-10-07-2, for alternative inspection.

55-617-00-02	AWL	55-05-02-211-803	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (Detailed) the exposed surface of the front spar upper chord between the upper skin and Leading Edge (LE) skin.
See Doc. D626A001 - DTR, DTR check form 55-10-07-2, for alternative inspection.

55-618-00-01	AWL	55-05-02-230-802	1.1	56000 FC	12000 FC	ALL	ALL

Inspect (Penetrant) the upper and lower Side of Body (SOB) spar bolts on the horizontal stabilizer front spar terminal fitting.
See Doc. D626A001 - DTR, DTR check form 55-10-07-3, for alternative inspection.

ACCESS NOTE: Removal of upper and lower spar bolts is required.

55-618-00-02	AWL	55-05-02-230-802	1.1	56000 FC	12000 FC	ALL	ALL

Inspect (Penetrant) the upper and lower Side of Body (SOB) spar bolts on the horizontal stabilizer front spar terminal fitting.
See Doc. D626A001 - DTR, DTR check form 55-10-07-3, for alternative inspection.

ACCESS NOTE: Removal of upper and lower spar bolts is required.

55-618-10-01	AWL	55-05-02-250-880	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High frequency Eddy Current) the lower chord at the end rib tension fitting installation bolt hole.
See Doc. D626A001 - DTR, DTR check form 55-10-07-4, for alternative inspection.

ACCESS NOTE: The inspection requires the removal of the bolt.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-618-10-02	AWL	55-05-02-250-880	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High frequency Eddy Current) the lower chord at the end rib tension fitting installation bolt hole.

See Doc. D626A001 - DTR, DTR check form 55-10-07-4, for alternative inspection.

ACCESS NOTE: The inspection requires the removal of the bolt.

55-619-00-01	AWL	55-05-02-250-816	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) all of the holes on the hinge plate, around the bore, at elevator STA 121 and STA 176.

See Doc. D626A001 - DTR, DTR check form 55-10-08-1, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-35.

ACCESS NOTE: Removal of hinge plate from assembly is required prior to inspection. The spherical bearing should not be removed.

55-619-00-02	AWL	55-05-02-250-816	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) all of the holes on the hinge plate, around the bore, at elevator STA 121 and STA 176.

See Doc. D626A001 - DTR, DTR check form 55-10-08-1, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-35.

ACCESS NOTE: Removal of hinge plate from assembly is required prior to inspection. The spherical bearing should not be removed.

55-620-00-01	AWL	55-05-02-250-817	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) all of the holes on the hinge plate, around the bore, at elevator STA 213 and STA 250.

See Doc. D626A001 - DTR, DTR check form 55-10-08-2, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-35.

ACCESS NOTE: Removal of hinge plate from assembly is required prior to inspection. The spherical bearing should not be removed.

55-620-00-02	AWL	55-05-02-250-817	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) all of the holes on the hinge plate, around the bore, at elevator STA 213 and STA 250.

See Doc. D626A001 - DTR, DTR check form 55-10-08-2, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-35.

ACCESS NOTE: Removal of hinge plate from assembly is required prior to inspection. The spherical bearing should not be removed.

55-621-00-01	AWL	55-05-02-250-818	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) all of the holes on the hinge plates around the bore at elevator STA 265.

See Doc. D626A001 - DTR, DTR check form 55-10-08-3, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-36.

ACCESS NOTE: Removal of hinge plate is required. The spherical bearing should not be removed.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-621-00-02	AWL	55-05-02-250-818	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) all of the holes on the hinge plates around the bore at elevator STA 265.

See Doc. D626A001 - DTR, DTR check form 55-10-08-3, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-36.

ACCESS NOTE: Removal of hinge plate is required. The spherical bearing should not be removed.

55-622-00-01	AWL	55-05-02-250-819	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) all holes on the hinge plate at elevator STA 66.

See Doc. D626A001 - DTR, DTR check form 55-10-08-4, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-36.

ACCESS NOTE: Removal of hinge plate is required. The spherical bearing should not be removed.

55-622-00-02	AWL	55-05-02-250-819	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) all holes on the hinge plate at elevator STA 66.

See Doc. D626A001 - DTR, DTR check form 55-10-08-4, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-36.

ACCESS NOTE: Removal of hinge plate is required. The spherical bearing should not be removed.

55-623-01-01	AWL	55-05-02-250-821	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) around the outer edge of the bushing flange on both the upper and lower trailing edge clevis lugs at elevator STA 121 and elevator STA 176.

See Doc. D626A001-DTR, DTR check form 55-10-08-5B, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-23.

AIRPLANE NOTE: This inspection applies to LN # 216 and on with hinge rib clevis lug thickness of t=0.25 inch.

55-623-01-02	AWL	55-05-02-250-821	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) around the outer edge of the bushing flange on both the upper and lower trailing edge clevis lugs at elevator STA 121 and elevator STA 176.

See Doc. D626A001-DTR, DTR check form 55-10-08-5B, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-23.

AIRPLANE NOTE: This inspection applies to LN # 216 and on with hinge rib clevis lug thickness of t=0.25 inch.

55-624-00-01	AWL	55-05-02-250-822	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) around the bushing flanges on both the upper and lower clevis lugs at elevator STA 66.

See Doc. D626A001-DTR, DTR check form 55-10-08-6, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-23.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-624-00-02	AWL	55-05-02-250-822	1.1	56000 FC	36000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) around the bushing flanges on both the upper and lower clevis lugs at elevator STA 66. See Doc. D626A001-DTR, DTR check form 55-10-08-6, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-23.						
55-625-00-01	AWL	55-05-02-250-823	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) around the bushing flanges at both the upper and lower clevis lugs at elevator STA 213 and elevator STA 250. See Doc. D626A001 - DTR, DTR check form 55-10-08-7, alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-23.						
55-625-00-02	AWL	55-05-02-250-823	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) around the bushing flanges at both the upper and lower clevis lugs at elevator STA 213 and elevator STA 250. See Doc. D626A001 - DTR, DTR check form 55-10-08-7, alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-23.						
55-626-00-01	AWL	55-05-02-250-824	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) around the edge of the bushing at both the upper and lower clevis lugs at elevator STA 265. Note: Bushing removal not required. See Doc. D626A001 - DTR, DTR check form 55-10-08-8, alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-23.						
55-626-00-02	AWL	55-05-02-250-824	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) around the edge of the bushing at both the upper and lower clevis lugs at elevator STA 265. Note: Bushing removal not required. See Doc. D626A001 - DTR, DTR check form 55-10-08-8, alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-23.						
55-627-00-01	AWL	55-05-02-250-825	1.1	56000 FC	12000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the elevator hinge rib chord around the fasteners common to the upper and lower spar chords at elevator STA 66 and elevator STA 121. See Doc. D626A001 - DTR, DTR check form 55-10-08-9, alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-37. **ACCESS NOTE:** Removal of lower composite skin panels is required.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-627-00-02	AWL	55-05-02-250-825	1.1	56000 FC	12000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the elevator hinge rib chord around the fasteners common to the upper and lower spar chords at elevator STA 66 and elevator STA 121.
See Doc. D626A001 - DTR, DTR check form 55-10-08-9, alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-37.

ACCESS NOTE: Removal of lower composite skin panels is required.

55-628-00-01	AWL	55-05-02-250-826	1.1	56000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the rib chord around the fasteners common to the stabilizer upper and lower rear spar chord at elevator STA 176.
See Doc. D626A001 - DTR, DTR check form 55-10-8-10, alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-37.

ACCESS NOTE: Removal of lower trailing edge panels is required.

55-628-00-02	AWL	55-05-02-250-826	1.1	56000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the rib chord around the fasteners common to the stabilizer upper and lower rear spar chord at elevator STA 176.
See Doc. D626A001 - DTR, DTR check form 55-10-8-10, alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-37.

ACCESS NOTE: Removal of lower trailing edge panels is required.

55-629-00-01	AWL	55-05-02-250-827	1.1	56000 FC	30000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the hinge rib to rear spar attachment, on both the upper and lower chords, at elevator STA 213, elevator STA 250 and elevator STA 265.
See Doc. D626A001 - DTR, DTR check form 55-10-8-11, alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-37.

ACCESS NOTE: Removal of elevator and cover panel is required. No fastener removal is required.

55-629-00-02	AWL	55-05-02-250-827	1.1	56000 FC	30000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the hinge rib to rear spar attachment, on both the upper and lower chords, at elevator STA 213, elevator STA 250 and elevator STA 265.
See Doc. D626A001 - DTR, DTR check form 55-10-8-11, alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-37.

ACCESS NOTE: Removal of elevator and cover panel is required. No fastener removal is required.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-630-00-01	AWL	55-05-02-250-828	1.1	56000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) around the fastener holes that are used to attach the composite skin panels to the rib and the upper chord and lower chord fastener locations between the rear spar and trailing edge beam at elevator STA 66 and elevator STA 121.

See Doc. D626A001 - DTR, DTR check form 55-10-8-13, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-33.

ACCESS NOTE: Removal of lower composite skin panels is required. No fastener removal required.

55-630-00-02	AWL	55-05-02-250-828	1.1	56000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) around the fastener holes that are used to attach the composite skin panels to the rib and the upper chord and lower chord fastener locations between the rear spar and trailing edge beam at elevator STA 66 and elevator STA 121.

See Doc. D626A001 - DTR, DTR check form 55-10-8-13, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-33.

ACCESS NOTE: Removal of lower composite skin panels is required. No fastener removal required.

55-631-00-01	AWL	55-05-02-250-829	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) both upper and lower rib chords around the fastener holes common to the trailing edge panels between the stabilizer rear spar and the trailing edge beam at elevator STA 176.

See Doc. D626A001 - DTR, DTR check form 55-10-8-14, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-33.

ACCESS NOTE: Removal of the lower trailing edge skin panels is required.

55-631-00-02	AWL	55-05-02-250-829	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) both upper and lower rib chords around the fastener holes common to the trailing edge panels between the stabilizer rear spar and the trailing edge beam at elevator STA 176.

See Doc. D626A001 - DTR, DTR check form 55-10-8-14, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-33.

ACCESS NOTE: Removal of the lower trailing edge skin panels is required.

55-631-10-01	AWL	55-05-02-250-809	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) around all fastener locations in the upper inspar skin between the SOB rib and the rib at stabilizer STA. 157.1.

See Doc. D626A001 - DTR, DTR check form 55-10-09-1, alternative inspection.

55-631-10-02	AWL	55-05-02-250-809	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) around all fastener locations in the upper inspar skin between the SOB rib and the rib at stabilizer STA. 157.1.

See Doc. D626A001 - DTR, DTR check form 55-10-09-1, alternative inspection.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-633-00-01	AWL	55-05-02-211-805	1.1	56000 FC	9000 FC	ALL	ALL
	Inspect (Detailed) the horizontal stabilizer lower inspar skin from side-of-body to the tip. See Doc. D626A001 - DTR, DTR check form 55-10-10-1, for alternative inspection.						
55-633-00-02	AWL	55-05-02-211-805	1.1	56000 FC	9000 FC	ALL	ALL
	Inspect (Detailed) the horizontal stabilizer lower inspar skin from side-of-body to the tip. See Doc. D626A001 - DTR, DTR check form 55-10-10-1, for alternative inspection.						
55-634-00-01	AWL	55-05-02-211-806	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (Detailed) all four (4) front spar upper side of body clevis lugs. See Doc. D626A001 - DTR, DTR check form 55-10-11-1, for alternative inspection. **ACCESS NOTE:** Removal of leading edge and gap seals is required.						
55-634-00-02	AWL	55-05-02-211-806	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (Detailed) all four (4) front spar upper side of body clevis lugs. See Doc. D626A001 - DTR, DTR check form 55-10-11-1, for alternative inspection. **ACCESS NOTE:** Removal of leading edge and gap seals is required.						
55-635-00-01	AWL	55-05-02-130-805	1.1	56000 FC	36000 FC	ALL	ALL
	Inspect (Ultrasonic) all four lower pivot fitting clevis lugs for hidden damage at the lug bore. See Doc. D626A001 - DTR, DTR check form 55-10-12-1, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 55-10-05. **ACCESS NOTE:** Removal of the gap covers and sliding seals is required.						
55-635-00-02	AWL	55-05-02-130-805	1.1	56000 FC	36000 FC	ALL	ALL
	Inspect (Ultrasonic) all four lower pivot fitting clevis lugs for hidden damage at the lug bore. See Doc. D626A001 - DTR, DTR check form 55-10-12-1, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 55-10-05. **ACCESS NOTE:** Removal of the gap covers and sliding seals is required.						
55-636-00-01	AWL	55-05-02-130-806	1.1	56000 FC	4000 FC	ALL	ALL
	Inspect (Ultrasonic) the side of body lug bore on the horizontal stabilizer center section rear spar upper chord. See Doc. D626A001 - DTR, DTR check form 55-10-12-2, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 55-10-08. **ACCESS NOTE:** Removal of gap covers is required.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-636-00-02	AWL	55-05-02-130-806	1.1	56000 FC	4000 FC	ALL	ALL

Inspect (Ultrasonic) the side of body lug bore on the horizontal stabilizer center section rear spar upper chord.
See Doc. D626A001 - DTR, DTR check form 55-10-12-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 55-10-08.

ACCESS NOTE: Removal of gap covers is required.

55-637-00-01	AWL	55-05-02-211-807	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (Detailed) the thrust beam to spar joint on the horizontal stabilizer center section front spar upper chord.
See Doc. D626A001 - DTR, DTR check form 55-10-13-2, for alternative inspection.

55-637-00-02	AWL	55-05-02-211-807	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (Detailed) the thrust beam to spar joint on the horizontal stabilizer center section front spar upper chord.
See Doc. D626A001 - DTR, DTR check form 55-10-13-2, for alternative inspection.

55-638-00-01	AWL	55-05-02-130-807	1.1	56000 FC	9000 FC	ALL	ALL

Inspect (Ultrasonic) the upper and lower chords on the primary beam at the front and rear spar joints.
See Doc. D626A001 - DTR, DTR check form 55-10-14-1, for alternative inspection.

ACCESS NOTE: Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

55-638-00-02	AWL	55-05-02-130-807	1.1	56000 FC	9000 FC	ALL	ALL

Inspect (Ultrasonic) the upper and lower chords on the primary beam at the front and rear spar joints.
See Doc. D626A001 - DTR, DTR check form 55-10-14-1, for alternative inspection.

ACCESS NOTE: Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

55-639-00-01	AWL	55-05-02-250-830	1.1	56000 FC	9000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the upper and lower chord splice members of the upper primary beam at the front and rear spar joint plates and angles.
See Doc. D626A001 - DTR, DTR check form 55-10-14-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-38.

ACCESS NOTE: Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

55-639-00-02	AWL	55-05-02-250-830	1.1	56000 FC	9000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the upper and lower chord splice members of the upper primary beam at the front and rear spar joint plates and angles.
See Doc. D626A001 - DTR, DTR check form 55-10-14-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-38.

ACCESS NOTE: Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-640-00-01	AWL	55-05-02-130-808	1.1	56000 FC	9000 FC	ALL	ALL

Inspect (Ultrasonic) the upper primary beam chord for damage under the thrust beam chord around fasteners common to the two chords and splice angle.
See Doc. D626A001 - DTR, DTR check form 55-10-14-3, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 55-10-14.

ACCESS NOTE: Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

55-640-00-02	AWL	55-05-02-130-808	1.1	56000 FC	9000 FC	ALL	ALL

Inspect (Ultrasonic) the upper primary beam chord for damage under the thrust beam chord around fasteners common to the two chords and splice angle.
See Doc. D626A001 - DTR, DTR check form 55-10-14-3, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 55-10-14.

ACCESS NOTE: Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

55-641-00-01	AWL	55-05-02-130-809	1.1	56000 FC	21000 FC	ALL	ALL

Inspect (Ultrasonic) the thrust beam upper and lower chords around the first row of fasteners common to the splice members at both the front and rear spar joints.
See Doc. D626A001 - DTR, DTR check form 55-10-14-4, for alternative inspection.

ACCESS NOTE: Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

55-641-00-02	AWL	55-05-02-130-809	1.1	56000 FC	21000 FC	ALL	ALL

Inspect (Ultrasonic) the thrust beam upper and lower chords around the first row of fasteners common to the splice members at both the front and rear spar joints.
See Doc. D626A001 - DTR, DTR check form 55-10-14-4, for alternative inspection.

ACCESS NOTE: Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

55-642-00-01	AWL	55-05-02-250-831	1.1	56000 FC	34000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the upper and lower thrust beam chords around the first row of fasteners common to the splice members at both the front and rear spars.
See Doc. D626A001 - DTR, DTR check form 55-10-14-5, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-39.

ACCESS NOTE: Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

55-642-00-02	AWL	55-05-02-250-831	1.1	56000 FC	34000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the upper and lower thrust beam chords around the first row of fasteners common to the splice members at both the front and rear spars.
See Doc. D626A001 - DTR, DTR check form 55-10-14-5, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-39.

ACCESS NOTE: Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-643-00-01	AWL	55-05-02-130-810	1.1	56000 FC	14000 FC	ALL	ALL

Inspect (Ultrasonic) upper and lower thrust beam chords forward and aft of the intersection joint (4 places).
See Doc. D626A001 - DTR, DTR check form 55-10-14-6, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 55-10-09.

ACCESS NOTE: Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

55-643-00-02	AWL	55-05-02-130-810	1.1	56000 FC	14000 FC	ALL	ALL

Inspect (Ultrasonic) upper and lower thrust beam chords forward and aft of the intersection joint (4 places).
See Doc. D626A001 - DTR, DTR check form 55-10-14-6, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 55-10-09.

ACCESS NOTE: Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

55-644-00-01	AWL	55-05-02-250-832	1.1	56000 FC	27000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) upper and lower thrust beam splice plates.
See Doc. D626A001 - DTR, DTR check form 55-10-14-7, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-20.

ACCESS NOTE: Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

55-644-00-02	AWL	55-05-02-250-832	1.1	56000 FC	27000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) upper and lower thrust beam splice plates.
See Doc. D626A001 - DTR, DTR check form 55-10-14-7, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-20.

ACCESS NOTE: Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

55-645-00-01	AWL	55-05-02-130-811	1.1	56000 FC	13000 FC	ALL	ALL

Inspect (Ultrasonic) the upper thrust beam chord for damage around the four fasteners common to the primary beam chord.
See Doc. D626A001 - DTR, DTR check form 55-10-14-8, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 55-10-10.

ACCESS NOTE: Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

55-645-00-02	AWL	55-05-02-130-811	1.1	56000 FC	13000 FC	ALL	ALL

Inspect (Ultrasonic) the upper thrust beam chord for damage around the four fasteners common to the primary beam chord.
See Doc. D626A001 - DTR, DTR check form 55-10-14-8, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 55-10-10.

ACCESS NOTE: Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-646-00-01	AWL	55-05-02-250-833	1.1	56000 FC	8000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the upper chord web flange around the fasteners common to the web between the front and rear spars from STA 1216 to STA 1242.
See Doc. D626A001 - DTR, DTR check form 55-10-14-9, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-25.

ACCESS NOTE: Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

55-646-00-02	AWL	55-05-02-250-833	1.1	56000 FC	8000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the upper chord web flange around the fasteners common to the web between the front and rear spars from STA 1216 to STA 1242.
See Doc. D626A001 - DTR, DTR check form 55-10-14-9, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-25.

ACCESS NOTE: Access horizontal stabilizer center section through opening in center of 1088 bulkhead.

55-647-00-01	AWL	55-05-02-250-834	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) all hinge fitting clevis lugs on the lug face around the circumference of the bushing.
See Doc. D626A001 - DTR, DTR check form 55-20-05, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-50-07.

55-647-00-02	AWL	55-05-02-250-834	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) all hinge fitting clevis lugs on the lug face around the circumference of the bushing.
See Doc. D626A001 - DTR, DTR check form 55-20-05, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-50-07.

55-648-00-01	AWL	55-05-02-250-835	1.1	56000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) each lug face of the elevator actuator fittings.
See Doc. D626A001 - DTR, DTR check form 55-20-06, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-50-09.

ACCESS NOTE: Remove the actuator rod assembly as required.

55-648-00-02	AWL	55-05-02-250-835	1.1	56000 FC	24000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) each lug face of the elevator actuator fittings.
See Doc. D626A001 - DTR, DTR check form 55-20-06, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-50-09.

ACCESS NOTE: Remove the actuator rod assembly as required.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-649-00-01	AWL	55-05-02-210-801	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (General Visual) the elevator horn balance weight support structure at elevator STA 260.
See Doc. D626A001 - DTR, DTR check form 55-20-07, for alternative inspection.

ACCESS NOTE: Removal of the elevator horn balance weight fairing (183A7400) is required.
Outboard rib is inspected through hole in inboard rib.

55-649-00-02	AWL	55-05-02-210-801	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (General Visual) the elevator horn balance weight support structure at elevator STA 260.
See Doc. D626A001 - DTR, DTR check form 55-20-07, for alternative inspection.

ACCESS NOTE: Removal of the elevator horn balance weight fairing (183A7400) is required.
Outboard rib is inspected through hole in inboard rib.

55-652-00-01	AWL	55-05-02-250-838	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) around the circumference of the washer at bolt hole 2 and 3 on the mast arm fitting upper flange.
See Doc. D626A001 - DTR, DTR check form 55-20-08-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-15.

AIRPLANE NOTE: Applicable to L/N 596 and ALL 1175 and on and L/N 1 to 1174 that have incorporated SB 737-55A1080 or SB 737-55-1081.

55-652-00-02	AWL	55-05-02-250-838	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) around the circumference of the washer at bolt hole 2 and 3 on the mast arm fitting upper flange.
See Doc. D626A001 - DTR, DTR check form 55-20-08-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-10-15.

AIRPLANE NOTE: Applicable to L/N 596 and ALL 1175 and on and L/N 1 to 1174 that have incorporated SB 737-55A1080 or SB 737-55-1081.

55-655-00-01	AWL	55-05-02-260-802	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (X-ray) the upper and lower flanges of hinges 3, 4, 5, and 6.
See Doc. D626A001 - DTR, DTR check form 55-20-09-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 2, Subject 55-10-01.

AIRPLANE NOTE: Applicable to L/N 596 and ALL 1175 and On, and L/N 1 - 1174 that have incorporated SB 737-55A1080 or SB 737-55-1081.

55-655-00-02	AWL	55-05-02-260-802	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (X-ray) the upper and lower flanges of hinges 3, 4, 5, and 6.
See Doc. D626A001 - DTR, DTR check form 55-20-09-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 2, Subject 55-10-01.

AIRPLANE NOTE: Applicable to L/N 596 and ALL 1175 and On, and L/N 1 - 1174 that have incorporated SB 737-55A1080 or SB 737-55-1081.

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-657-00-01	AWL	55-05-02-250-841	1.1	56000 FC	36000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the surface of the fitting and doublers of each bracket assembly around the lug bore. See Doc. D626A001 - DTR, DTR check form 55-20-12-1, for alternative inspection. **AIRPLANE NOTE:** Applicable to L/N 596 and ALL 1175 and on, and L/N 1 - 1174 that have incorporated SB 737-55A1080 or SB 737-55-1081. **ACCESS NOTE:** Remove upper or lower horizontal stabilizer trailing edge seal and elevator inboard hinge cover panel. Bushing removal is not required.						
55-657-00-02	AWL	55-05-02-250-841	1.1	56000 FC	36000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the surface of the fitting and doublers of each bracket assembly around the lug bore. See Doc. D626A001 - DTR, DTR check form 55-20-12-1, for alternative inspection. **AIRPLANE NOTE:** Applicable to L/N 596 and ALL 1175 and on, and L/N 1 - 1174 that have incorporated SB 737-55A1080 or SB 737-55-1081. **ACCESS NOTE:** Remove upper or lower horizontal stabilizer trailing edge seal and elevator inboard hinge cover panel. Bushing removal is not required.						
55-658-00-01	AWL	55-05-02-250-842	1.1	56000 FC	24000 FC	600 700 800 900 900ER	ALL
	Inspect (High Frequency Eddy Current) the exposed chord between the inspar and trailing edge skin from vertical fin STA 73.4 to vertical fin tip. See Doc. D626A001 - DTR, DTR check form 55-30-03-2, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-30-05.						
55-658-00-02	AWL	55-05-02-250-842	1.1	56000 FC	24000 FC	600 700 800 900 900ER	ALL
	Inspect (High Frequency Eddy Current) the exposed chord between the inspar and trailing edge skin from vertical fin STA 73.4 to vertical fin tip. See Doc. D626A001 - DTR, DTR check form 55-30-03-2, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 55-30-05.						
55-660-00-01	AWL	55-05-02-211-808	1.1	56000 FC	36000 FC	ALL	ALL
	Inspect (Detailed) the chords lower flange and land-up at the forward end fastener location common to rear spar terminal fitting. See Doc. D626A001 - DTR, DTR check form 55-30-05-1, for alternative inspection.						
55-660-00-02	AWL	55-05-02-211-808	1.1	56000 FC	36000 FC	ALL	ALL
	Inspect (Detailed) the chords lower flange and land-up at the forward end fastener location common to rear spar terminal fitting. See Doc. D626A001 - DTR, DTR check form 55-30-05-1, for alternative inspection.						
55-662-00-01	AWL	55-05-02-211-809	1.1	56000 FC	24000 FC	ALL	ALL
	Inspect (Detailed) the hinge fitting lugs at the rib attachment for hinges #1, #2, #3, #4, #5, #6, #7, #7A and #8. See Doc. D626A001 - DTR, DTR check form 55-30-06-1, for alternative inspection.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-663-00-01	AWL	55-05-02-211-810	1.1	56000 FC	36000 FC	ALL	ALL
	Inspect (Detailed) rudder hinge ribs 1 through 7A at attachment to both rear spar chords. See Doc. D626A001 - DTR, DTR check form 55-30-06-2, for alternative inspection. **ACCESS NOTE:** Removal of the skin panels aft of the fin spar is required.						
55-664-00-01	AWL	55-05-02-211-811	1.1	56000 FC	9000 FC	ALL	ALL
	Inspect (Detailed) all the rudder hinge fitting clevis lugs and the hinge fitting attachments to the skin and the spar. See Doc. D626A001 - DTR, DTR check form 55-40-05, for alternative inspection.						
55-665-00-01	AWL	55-05-02-250-844	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the rudder actuator fitting clevis lugs at rudder STA 60.85 and rudder STA 70.65, as well as visually inspect the hinge fitting attachments to the skin and the spar. See Doc. D626A001 - DTR, DTR check form 55-40-06, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 55-50-08. **ACCESS NOTE:** Removal of the actuator rod for access to the inner lug faces is required.						
55-666-00-01	AWL	55-05-02-211-812	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (Detailed) the casting in the area around the doubler as well as the balance arm assembly attachment to the rudder front spar. See Doc. D626A001 - DTR, DTR check form 55-40-07, for alternative inspection.						
55-800-00-01	MRB	05-41-03-210-806	1.1	120 DY	120 DY	ALL	ALL
	Perform an external zonal inspection (GV) of the vertical fin and horizontal stabilizer. Inspection is accomplished from the ground, without the use of stands or ladders. No additional access panel required.						
55-802-00-01	MRB	05-41-03-210-807	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the vertical fin - dorsal fin. **INTERVAL NOTE:** Whichever comes first.						
55-804-00-01	MRB	05-41-03-210-808	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the vertical fin - leading edge (only if HF system installed). **AIRPLANE NOTE:** Only if HF system installed. **INTERVAL NOTE:** Whichever comes first.						
55-806-00-01	MRB	05-41-03-210-809	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the vertical fin - leading edge. **INTERVAL NOTE:** Whichever comes first.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-808-00-01	MRB	05-41-03-210-810	1.1 1.2	36000 FC 12 YR	36000 FC 12 YR	ALL	ALL
	Perform an internal zonal inspection (GV) of the vertical fin - leading edge. **INTERVAL NOTE:** Whichever comes first.						
55-810-00-01	MRB	05-41-03-210-811	1.1 1.2	4800 FC 24 MO	4800 FC 24 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the vertical fin - front spar to rear spar. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 36000 FC/12 YR is satisfied by this zonal inspection.						
55-812-00-01	MRB	05-41-03-210-812	1.1 1.2	5500 FC 30 MO	5500 FC 30 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the vertical fin - front spar to rear spar. **INTERVAL NOTE:** Whichever comes first.						
55-814-00-01	MRB	05-41-03-210-813	1.1 1.2	4800 FC 24 MO	4800 FC 24 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the vertical fin - rear spar to trailing edge. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 36000 FC/12 YR is satisfied by this zonal inspection.						
55-816-00-01	MRB	05-41-03-210-814	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the vertical fin - rear spar to trailing edge. **INTERVAL NOTE:** Whichever comes first.						
55-818-00-01	MRB	05-41-03-210-815	1.1 1.2	5500 FC 30 MO	5500 FC 30 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the rudder. **INTERVAL NOTE:** Whichever comes first.						
55-820-00-01	MRB	05-41-03-210-816	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the vertical fin tip. **INTERVAL NOTE:** Whichever comes first.						
55-822-01-01	MRB	05-41-03-210-817	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the horizontal stabilizer - leading edge - left. **INTERVAL NOTE:** Whichever comes first.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-824-01-01	MRB	05-41-03-210-818	1.1 1.2	4000 FC 18 MO	4000 FC 18 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the horizontal stabilizer - front spar to rear spar - left. **INTERVAL NOTE:** Whichever comes first.						
55-826-01-01	MRB	05-41-03-210-819	1.1 1.2	4000 FC 18 MO	4000 FC 18 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the inboard horizontal stabilizer - front spar to rear spar - left. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** As an alternate to the access provided, access panel 332AB can be removed for inspection from lower side of stabilizer.						
55-828-01-01	MRB	05-41-03-210-820	1.1 1.2	4800 FC 24 MO	4800 FC 24 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the horizontal stabilizer - rear spar to trailing edge - left. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 36000 FC/12 YR is satisfied by this zonal inspection.						
55-830-01-01	MRB	05-41-03-210-821	1.1 1.2	4800 FC 24 MO	4800 FC 24 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the horizontal stabilizer - rear spar to trailing edge - left. **INTERVAL NOTE:** Whichever comes first.						
55-832-01-01	MRB	05-41-03-210-822	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the horizontal stabilizer - elevator - left. **INTERVAL NOTE:** Whichever comes first.						
55-834-01-01	MRB	05-41-03-210-823	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the horizontal stabilizer - stabilizer tip -left. **INTERVAL NOTE:** Whichever comes first.						
55-836-02-01	MRB	05-41-03-210-824	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the horizontal stabilizer - leading edge - right. **INTERVAL NOTE:** Whichever comes first.						
55-838-02-01	MRB	05-41-03-210-825	1.1 1.2	4000 FC 18 MO	4000 FC 18 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the horizontal stabilizer - front spar to rear spar - right. **INTERVAL NOTE:** Whichever comes first.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
55-840-02-01	MRB	05-41-03-210-826	1.1 1.2	4000 FC 18 MO	4000 FC 18 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the inboard horizontal stabilizer - front spar to rear spar - right. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** With access provided. Access panel 342AB can be used as an alternate to inspect from the lower side of the stabilizer.						
55-842-02-01	MRB	05-41-03-210-827	1.1 1.2	4800 FC 24 MO	4800 FC 24 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the horizontal stabilizer - rear spar to trailing edge - right. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 36000 FC/12 YR is satisfied by this zonal inspection.						
55-844-02-01	MRB	05-41-03-210-828	1.1 1.2	4800 FC 24 MO	4800 FC 24 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the horizontal stabilizer - rear spar to trailing edge - right. **INTERVAL NOTE:** Whichever comes first.						
55-846-02-01	MRB	05-41-03-210-829	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the horizontal stabilizer - elevator - right. **INTERVAL NOTE:** Whichever comes first.						
55-848-02-01	MRB	05-41-03-210-830	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the horizontal stabilizer - stabilizer tip - right. **INTERVAL NOTE:** Whichever comes first.						
56-010-00-01	MRB	12-25-81-600-801	1.1	2 YR	2 YR	600 700 800 900	ALL
	Lubricate the inside release mechanisms for the pilots and co-pilots #2 sliding windows. Lubricate the outside release mechanism for the co-pilots #2 sliding window.						
56-030-00-01	MRB	56-12-11-710-803	1.1	4 YR	4 YR	600 700 800 900	ALL
	Operationally check the inside release mechanisms for the pilots and co-pilots #2 sliding windows. Operationally check the co-pilots #2 sliding window from the outside.						
56-050-00-01	MRB	56-12-11-200-802	1.1	4 YR	4 YR	600 700 800 900	ALL
	Perform a general visual inspection of the pilots and co-pilots #2 sliding window sill drain for obvious damage, clogging, condition, and security. **AIRPLANE NOTE:** Applicable to airplanes line number 145 and on. Applicable to airplanes line number 1 to 144 incorporating SB 737-56-1011.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-010-00-01	MRB	51-05-01-210-804 57-05-03-210-801	1.1 1.2	10 YR 36000 FC	10 YR 36000 FC	ALL	ALL
	Inspect forward side of wing center section front spar, including the side of body/terminal fitting. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Remove aft panels in forward cargo compartment.						
57-020-00-01	MRB	51-05-01-210-809 57-05-03-210-802	1.1 1.2	10 YR 36000 FC	10 YR 36000 FC	ALL	ALL
	Inspect inside wing center section: 1. Upper surface (including skins, typical and vent stringers, splice stringer, at attachment to floor beams); 2. Lower surface (including skins, typical stringers, splice stringers, at attachment to keel beam, at drain installation, at access hole, at attachment to fuselage drag angle, at attachment to lower beam at BL 41); 3. Front and rear spars (including webs and stiffeners, upper and lower spar chords, attachments to skin); 4. Side of body rib (including webs and stiffeners, upper rib chord, lower tee chord, front and rear spar terminal fittings, splice fittings); 5. Spanwise beams. **INTERVAL NOTE:** Whichever comes first.						
57-030-00-01	MRB	51-05-01-210-806 57-05-03-210-803	1.1	9 YR	3 YR	ALL	ALL
	Inspect aft side of rear spar, including keel beam stiffeners at BL 6.2, and side of body rear spar terminal fitting.						
57-040-00-01	MRB	51-05-01-210-804 57-05-03-210-804	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL
	Inspect lower side of lower surface of wing center section: 1. Skins, typical stringers, splice stringers, front and rear spar lower chords; 2. Side of body lower tee chord and splice plates; 3. Lower beams at BL 41; 4. At attachments to keel beam, to lower beams at BL 41, to fuselage drag angles; at drain installation and access holes. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Remove ECS heat exchanger access panel.						
57-050-00-01	MRB	51-05-01-210-806 57-05-03-210-805	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL
	Inspect forward side of front spar, including side of body/terminal fitting attachments. **INTERVAL NOTE:** Whichever comes first.						
57-060-01-01	MRB	51-05-01-210-806 57-05-03-210-806	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL
	Inspect left outboard wing lower surface (under lower side of body fairing), including attachment locations. **INTERVAL NOTE:** Whichever comes first.						
57-060-02-01	MRB	51-05-01-210-806 57-05-03-210-807	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL
	Inspect right outboard wing lower surface (under lower side of body fairing), including attachment locations. **INTERVAL NOTE:** Whichever comes first.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-070-01-01	MRB	51-05-01-210-806 57-05-03-210-808	1.1 1.2	9 YR 18000 FC	9 YR 18000 FC	ALL	ALL
	Inspect upper side of left outboard wing upper surface (under side-of-body fairing), including: 1. Wing upper surface at side-of-body splice, including upper rib chord; 2. Wing upper surface at attachment locations. **INTERVAL NOTE:** Whichever comes first.						
57-070-02-01	MRB	51-05-01-210-806 57-05-03-210-809	1.1 1.2	9 YR 18000 FC	9 YR 18000 FC	ALL	ALL
	Inspect upper side of right outboard wing upper surface (under side-of-body fairing), including: 1. Wing upper surface at side-of-body splice, including upper rib chord; 2. Wing upper surface at attachment locations. **INTERVAL NOTE:** Whichever comes first.						
57-090-00-01	MRB	51-05-01-210-808 57-05-03-210-810	1.1 1.2	10 YR 36000 FC	10 YR 36000 FC	ALL	ALL
	Inspect upper side of upper wing surface: 1. Upper panel, including at attachments, front and rear spar upper chords, and side of body upper rib chord; 2. Floor beams from Sta 540 to Sta 727, including at floor beam attachments; 3. Inspect for condition of secondary vapor barrier. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Remove floor panels and insulation as required in passenger compartment as for access.						
57-100-01-01	MRB	51-05-01-210-801 57-05-03-210-811	1.1 1.2	48 MO 9000 FC	48 MO 9000 FC	ALL	ALL
	Inspect the following fittings: 1. Front spar pitch load fitting (R1); 2. Aft drag load fitting (R2); 3. Outboard side load fitting (R3); 4. Inboard side load fitting (R4); 5. Side brace support fittings (R7 and R8); 6. R3 forward backup fitting. Inspect upper wing skin at attachment to nacelle fitting R1. **INTERVAL NOTE:** Whichever comes first.						
57-100-02-01	MRB	51-05-01-210-801 57-05-03-210-812	1.1 1.2	48 MO 9000 FC	48 MO 9000 FC	ALL	ALL
	Inspect the following fittings: 1. Front spar pitch load fitting (R1); 2. Aft drag load fitting (R2); 3. Outboard side load fitting (R3); 4. Inboard side load fitting (R4); 5. Side brace support fittings (R7 and R8); 6. R3 forward backup fitting. Inspect upper wing skin at attachment to nacelle fitting R1. **INTERVAL NOTE:** Whichever comes first.						
57-110-01-01	MRB	51-05-01-210-809 57-05-03-210-813	1.1 1.2	48 MO 9000 FC	48 MO 9000 FC	ALL	ALL
	Inspect left wing lower surface under strut fairing, including all attachment locations. **INTERVAL NOTE:** Whichever comes first.						
57-110-02-01	MRB	51-05-01-210-809 57-05-03-210-814	1.1 1.2	48 MO 9000 FC	48 MO 9000 FC	ALL	ALL
	Inspect right wing lower surface under strut fairing, including all attachment locations. **INTERVAL NOTE:** Whichever comes first.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-120-01-01	MRB	51-05-01-210-806 57-05-03-210-815	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL
	Inspect left front spar chords, webs and stiffeners, including at side of body joint and at nacelle fitting attachment. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Deploy Krueger Flaps.						
57-120-02-01	MRB	51-05-01-210-806 57-05-03-210-816	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL
	Inspect right front spar chords, webs and stiffeners, including at side of body joint and at nacelle fitting attachment. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Deploy Krueger Flaps.						
57-130-01-01	MRB	51-05-01-210-806 57-05-03-210-817	1.1	10 YR	10 YR	ALL	ALL
	Inspect left wing leading edge cavity, including flaps and slats. **ACCESS NOTE:** Extend Krueger flaps and slats.						
57-130-02-01	MRB	51-05-01-210-806 57-05-03-210-818	1.1	10 YR	10 YR	ALL	ALL
	Inspect right wing leading edge cavity, including flaps and slats. **ACCESS NOTE:** Extend Krueger flaps and slats.						
57-140-01-01	MRB	51-05-01-210-809 57-05-03-211-801	1.1	6 YR	6 YR	ALL	ALL
	Inspect left wing slat tracks. **ACCESS NOTE:** Extend slats.						
57-140-02-01	MRB	51-05-01-210-809 57-05-03-211-802	1.1	6 YR	6 YR	ALL	ALL
	Inspect right wing slat tracks. **ACCESS NOTE:** Extend stats.						
57-160-01-01	MRB	51-05-01-210-809 57-05-03-210-819	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL
	Inspect fuel access holes in left outboard wing lower surface. (Tank entry is not required.) **INTERVAL NOTE:** Whichever comes first.						
57-160-02-01	MRB	51-05-01-210-809 57-05-03-210-820	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL
	Inspect fuel access holes in right outboard wing lower surface. (Tank entry is not required.) **INTERVAL NOTE:** Whichever comes first.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-170-01-01	MRB	51-05-01-210-809 57-05-03-210-821	1.1 1.2	10 YR 36000 FC	10 YR 36000 FC	ALL	ALL

Inspect inside left outboard wing from side of body Rib to Rib 5:

1. Side of body rib (including webs and stiffeners, upper rib chord, lower tee chord, splice fittings, terminal fittings at front and rear spars); 2. Upper and lower surfaces (including skins; typical splice, vent and rail stringers; at drain installations; at attachments to front and rear spars; at attachments to shear tied ribs and support fittings); 3. Front and rear spar chords, webs, stiffeners and rib posts; 4. Shear tied and non-shear tied ribs.

INTERVAL NOTE: Whichever comes first.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-170-02-01	MRB	51-05-01-210-809 57-05-03-210-822	1.1 1.2	10 YR 36000 FC	10 YR 36000 FC	ALL	ALL

Inspect inside right outboard wing from side of body Rib to Rib 5:

1. Side of body rib (including webs and stiffeners, upper rib chord, lower tee chord, splice fittings, terminal fittings at front and rear spars); 2. Upper and lower surfaces (including skins; typical splice, vent and rail stringers; at drain installations; at attachments to front and rear spars; at attachments to shear tied ribs and support fittings); 3. Front and rear spar chords, webs, stiffeners and rib posts; 4. Shear tied and non-shear tied ribs.

INTERVAL NOTE: Whichever comes first.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-180-01-01	MRB	51-05-01-210-809 57-05-03-210-823	1.1 1.2	10 YR 36000 FC	10 YR 36000 FC	ALL	ALL

Inspect inside left outboard wing from Rib 5 to Rib 22:

1. Upper and lower surfaces (including skins; typical splice, vent and rail stringers; at attachments to front and rear spars; at attachments to shear tied ribs and support fittings; at attachment to nacelle fittings; at drain installations); 2. Front spar chords, webs, stiffeners and rib posts, including at nacelle fitting attachments; 3. Rear spar chords, webs, stiffeners and rib posts, flap track support fittings, including rear spar at major fitting attachments; 4. Shear tied and non-shear tied ribs (including Rib 6 structural doors, Ribs 6 and 7 at nacelle support fittings, Ribs 10 and 14 at flap track support fittings); 5. Nacelle support fittings (R2 backup link, R4 back up link, R7/8 backup fitting).

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Do not remove 532AZ and 532BZ access doors at the same time.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-180-02-01	MRB	51-05-01-210-809 57-05-03-210-824	1.1 1.2	10 YR 36000 FC	10 YR 36000 FC	ALL	ALL

Inspect inside right outboard wing from Rib 5 to Rib 22:

1. Upper and lower surfaces (including skins; typical splice, vent and rail stringers; at attachments to front and rear spars; at attachments to shear tied ribs and support fittings; at attachment to nacelle fittings; at drain installations); 2. Front spar chords, webs, stiffeners and rib posts, including at nacelle fitting attachments; 3. Rear spar chords, webs, stiffeners and rib posts, flap track support fittings, including rear spar at major fitting attachments; 4. Shear tied and non-shear tied ribs (including Rib 6 structural doors, Ribs 6 and 7 at nacelle support fittings, Ribs 10 and 14 at flap track support fittings); 5. Nacelle support fittings (R2 backup link, R4 back up link, R7/8 backup fitting).

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Do not remove the 632AZ and 632BZ access doors at the same time.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-190-01-01	MRB	51-05-01-210-806 57-05-03-210-825	1.1 1.2	10 YR 36000 FC	10 YR 36000 FC	ALL	ALL

Inspect inside left outboard wing surge tank and dry bay (from Rib 22 to Rib 27), including upper and lower skins, stringers, front and rear spars, rib posts, WBL 656.17 closure rib (if provisioned for winglets, L/N 778 and on), shear tied and non-shear tied ribs, and access holes.

INTERVAL NOTE: Whichever comes first.

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-190-02-01	MRB	51-05-01-210-806 57-05-03-210-826	1.1 1.2	10 YR 36000 FC	10 YR 36000 FC	ALL	ALL

Inspect inside right outboard wing surge tank and dry bay (from Rib 22 to Rib 27), including upper and lower skins, stringers, front and rear spars, rib posts, WBL 656.17 closure rib (if provisioned for winglets, L/N 778 and on), shear tied and non-shear tied ribs, and access holes.

INTERVAL NOTE: Whichever comes first.

57-200-01-01	MRB	51-05-01-210-802 57-05-03-210-827	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL

Inspect left inboard flap inboard track assembly, carriage assembly, forward fitting assembly, aft link and aft link pins. Normal overhaul procedures, applied with the flap track assemblies, carriage assemblies and forward fitting assemblies removed, at intervals not exceeding 10 years, are adequate to maintain corrosion at safe levels on these components. Therefore application of the basic tasks and reporting are not required on these components.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Remove flap drive lube access door and access door from MLG wheel well.

57-200-02-01	MRB	51-05-01-210-802 57-05-03-210-828	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL

Inspect right inboard flap inboard track assembly, carriage assembly, forward fitting assembly, aft link and aft link pins. Normal overhaul procedures, applied with the flap track assemblies, carriage assemblies and forward fitting assemblies removed, at intervals not exceeding 10 years, are adequate to maintain corrosion at safe levels on these components. Therefore application of the basic tasks and reporting are not required on these components.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Remove flap drive lube access door and access door from MLG wheel well.

57-210-01-01	MRB	51-05-01-210-802 57-05-03-210-829	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL

Inspect left inboard flap outboard track assembly, carriage assembly, forward fitting assembly, and aft attach fitting. Normal overhaul procedures, applied with the flap track assemblies, carriage assemblies and forward fitting assemblies removed, at intervals not exceeding 10 years, are adequate to maintain corrosion at safe levels on these components. Therefore application of the basic tasks and reporting are not required on these components.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Remove flap support forward fairing and deploy flaps.

57-210-02-01	MRB	51-05-01-210-802 57-05-03-210-830	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL

Inspect right inboard flap outboard track assembly, carriage assembly, forward fitting assembly, and aft attach fitting. Normal overhaul procedures, applied with the flap track assemblies, carriage assemblies and forward fitting assemblies removed, at intervals not exceeding 10 years, are adequate to maintain corrosion at safe levels on these components. Therefore application of the basic tasks and reporting are not required on these components.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Remove flap support forward fairing and deploy flaps.

57-220-01-01	MRB	51-05-01-210-801 57-05-03-210-831	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL

Inspect lower side of lower surface (under flap support No. 3 fairing), including all attachment locations and access holes.

INTERVAL NOTE: Whichever comes first.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-220-02-01	MRB	51-05-01-210-801 57-05-03-210-832	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL
	Inspect lower side of lower surface (under flap support No. 6 fairing), including all attachment locations and access holes. **INTERVAL NOTE:** Whichever comes first.						
57-230-01-01	MRB	51-05-01-210-803 57-05-03-210-833	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL
	Inspect left outboard flap inboard and outboard track assemblies, carriage assemblies, forward fitting assemblies, and aft links. Normal overhaul procedures, applied with the flap track assemblies, carriage assemblies and forward fitting assemblies removed, at intervals not exceeding 10 years, are adequate to maintain corrosion at safe levels on these components. Therefore application of the basic tasks and reporting are not required on these components. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Remove flap forward fairings and deploy flaps.						
57-230-02-01	MRB	51-05-01-210-803 57-05-03-210-834	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL
	Inspect right outboard flap inboard and outboard track assemblies, carriage assemblies, forward fitting assemblies, and aft links. Normal overhaul procedures, applied with the flap track assemblies, carriage assemblies and forward fitting assemblies removed, at intervals not exceeding 10 years, are adequate to maintain corrosion at safe levels on these components. Therefore application of the basic tasks and reporting are not required on these components. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Remove flap forward fairings and deploy flaps.						
57-240-01-01	MRB	51-05-01-210-801 57-05-03-210-835	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL
	Inspect lower side of lower surface (under flap support No. 1 & 2 fairings), including all attachment locations and access holes. **INTERVAL NOTE:** Whichever comes first.						
57-240-02-01	MRB	51-05-01-210-801 57-05-03-210-836	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL
	Inspect lower side of lower surface (under flap support No. 7 & 8 fairings), including all attachment locations and access holes. **INTERVAL NOTE:** Whichever comes first.						
57-250-01-01	MRB	51-05-01-210-806 57-05-03-210-837	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL
	Inspect left main landing gear support structure: 1. Main landing gear beam assembly; 2. Outboard support (dog house) assembly; 3. Inboard support (hanger link) assembly; 4. Trunnion support assembly; 5. Stabilizer links, including attach fittings and fuse pins. **INTERVAL NOTE:** Whichever comes first.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-250-02-01	MRB	51-05-01-210-806 57-05-03-210-838	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL
	Inspect right main landing gear support structure: 1. Main landing gear beam assembly; 2. Outboard support (dog house) assembly; 3. Inboard support (hanger link) assembly; 4. Trunnion support assembly; 5. Stabilizer links, including attach fittings and fuse pins. **INTERVAL NOTE:** Whichever comes first.						
57-260-01-01	MRB	51-05-01-210-806 57-05-03-210-839	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL
	Inspect aft side of rear spar (chords, webs and stiffeners), including at main landing gear outboard support attachment, and at trunnion attachment. **INTERVAL NOTE:** Whichever comes first.						
57-260-02-01	MRB	51-05-01-210-806 57-05-03-210-840	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL
	Inspect aft side of rear spar (chords, webs and stiffeners), including at main landing gear outboard support attachment, and at trunnion attachment. **INTERVAL NOTE:** Whichever comes first.						
57-270-01-01	MRB	51-05-01-210-806 57-05-03-210-841	1.1	10 YR	5 YR	ALL	ALL
	Inspect the interior of left wing trailing edge cavity, including skins, ribs, ailerons and spoilers. **ACCESS NOTE:** Flap extension required for inspection.						
57-270-02-01	MRB	51-05-01-210-806 57-05-03-210-842	1.1	10 YR	5 YR	ALL	ALL
	Inspect the interior of right wing trailing edge cavity, including skins, ribs, ailerons and spoilers. **ACCESS NOTE:** Flap extension required for inspection.						
57-280-01-01	MRB	51-05-01-210-801 57-05-03-210-843	1.1 1.2	9 YR 18000 FC	9 YR 18000 FC	ALL	ALL
	Inspect left inboard ground spoiler actuator fittings. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Extend flaps and ground spoilers.						
57-280-02-01	MRB	51-05-01-210-801 57-05-03-210-844	1.1 1.2	9 YR 18000 FC	9 YR 18000 FC	ALL	ALL
	Inspect right inboard ground spoiler actuator fittings. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Extend flaps and ground spoilers.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-300-01-01	MRB	51-05-01-210-806 57-05-03-210-845	1.1 1.2	10 YR 36000 FC	10 YR 36000 FC	ALL	ALL
	Inspect left inboard flap internally: 1. Front spar (aft side), rear spar (forward side), inspar ribs, torque tube, torque tube ribs; 2. Aft flap track support assembly attachment on main flap rear spar. **INTERVAL NOTE:** Whichever comes first.						
57-300-02-01	MRB	51-05-01-210-806 57-05-03-210-846	1.1 1.2	10 YR 36000 FC	10 YR 36000 FC	ALL	ALL
	Inspect right inboard flap internally: 1. Front spar (aft side), rear spar (forward side), inspar ribs, torque tube, torque tube ribs; 2. Aft flap track support assembly attachment on main flap rear spar. **INTERVAL NOTE:** Whichever comes first.						
57-310-01-01	MRB	51-05-01-210-806 57-05-03-210-847	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL
	Inspect aft side of rear spar (chords, webs and stiffeners), including flap track 1 & 2 support fittings. **INTERVAL NOTE:** Whichever comes first.						
57-310-02-01	MRB	51-05-01-210-806 57-05-03-210-848	1.1 1.2	6 YR 18000 FC	6 YR 18000 FC	ALL	ALL
	Inspect aft side of rear spar (chords, webs and stiffeners), including flap track 7 & 8 support fittings. **INTERVAL NOTE:** Whichever comes first.						
57-330-01-01	MRB	51-05-01-210-806 57-05-03-210-849	1.1 1.2	10 YR 36000 FC	10 YR 36000 FC	ALL	ALL
	Inspect left outboard flap internally: 1. Front spar (aft side), including support fittings at WBL 254 and 358; 2. Rear spar (forward side); 3. Inspar ribs and aft flap track support ribs. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Deploy aft flap so that aft flap tracks do not block view of rear spar lower chord.						
57-330-02-01	MRB	51-05-01-210-806 57-05-03-210-850	1.1 1.2	10 YR 36000 FC	10 YR 36000 FC	ALL	ALL
	Inspect right outboard flap internally: 1. Front spar (aft side), including support fittings at WBL 254 and 358; 2. Rear spar (forward side); 3. Inspar ribs and aft flap track support ribs. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Deploy aft flap so that aft flap tracks do not block view of rear spar lower chord.						
57-340-01-01	MRB	51-05-01-210-806 57-05-03-211-803	1.1	6 YR	6 YR	ALL	ALL
	Inspect aluminum rib structure at winglet stations 0, 1, and 4. Utilize borescope to inspect the flanges adjacent to skin panels and spars. Note: This task satisfies the requirement of the Airplane Partners Boeing (APB) task 57-340-01. **AIRPLANE NOTE:** All airplanes equipped with winglets. **ACCESS NOTE:** Access through cover 527AB.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-340-02-01	MRB	51-05-01-210-806 57-05-03-211-804	1.1	6 YR	6 YR	ALL	ALL
	Inspect aluminum rib structure at winglet stations 0, 1, and 4. Utilize borescope to inspect the flanges adjacent to skin panels and spars. Note: This task satisfies the requirement of the Airplane Partners Boeing (APB) task 57-340-02. **AIRPLANE NOTE:** All airplanes equipped with winglets. **ACCESS NOTE:** Access through cover 627AB.						
57-351-01-01	MRB	51-05-01-210-804 57-05-03-211-805	1.1	6 YR	6 YR	ALL	ALL
	Inspect upper and lower flanges and webs, including barrel nut holes, Winglet Rib 0 and Wing Rib 27, WBL 658.17. Note: This task satisfies the requirement of the Airplane Partners Boeing (APB) task 57-350-01. **AIRPLANE NOTE:** All airplanes equipped with winglets. **ACCESS NOTE:** Remove winglet assembly. Remove winglet access panels as noted. Remove barrel nuts to facilitate inspection of recesses.						
57-351-02-01	MRB	51-05-01-210-804 57-05-03-211-806	1.1	6 YR	6 YR	ALL	ALL
	Inspect upper and lower flanges and webs, including barrel nut holes, Winglet Rib 0 and Wing Rib 27, WBL 658.17. Note: This task satisfies the requirement of the Airplane Partners Boeing (APB) task 57-350-02. **AIRPLANE NOTE:** All airplanes equipped with winglets. **ACCESS NOTE:** Remove winglet assembly. Remove winglet access panels as noted. Remove barrel nuts to facilitate inspection of recesses.						
57-600-00-01	AWL	57-05-02-130-801	1.1	56000 FC	36000 FC	ALL	ALL
	Inspect (Ultrasonic) the upper side of body splice at the double plus chord/stub beam interface at STA 639. See Doc. D626A001 - DTR, DTR check form 57-10-05-1, for alternative inspection. **ACCESS NOTE:** Inspection requires removal of wing-to-body fairing.						
57-600-00-02	AWL	57-05-02-130-801	1.1	56000 FC	36000 FC	ALL	ALL
	Inspect (Ultrasonic) the upper side of body splice at the double plus chord/stub beam interface at STA 639. See Doc. D626A001 - DTR, DTR check form 57-10-05-1, for alternative inspection. **ACCESS NOTE:** Inspection requires removal of wing-to-body fairing.						
57-601-00-01	AWL	57-05-02-250-801	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the double plus chord at stringer 18-A interface located at aft end of upper vertical flange radius at vertical flange/horizontal flange. See Doc. D626A001 - DTR, DTR check form 57-10-05-2, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-10-84. **ACCESS NOTE:** Inspection requires removal of wing-to-body fairing.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-601-00-02	AWL	57-05-02-250-801	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the double plus chord at stringer 18-A interface located at aft end of upper vertical flange radius at vertical flange/horizontal flange.
See Doc. D626A001 - DTR, DTR check form 57-10-05-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-10-84.

ACCESS NOTE: Inspection requires removal of wing-to-body fairing.

57-601-10-01	AWL	57-05-02-130-802	1.1	56000 FC	18000 FC	600 700 700C 700IGW 800 900	ALL

Inspect (Ultrasonic) the double plus chord vertical flange at body stringer 18A interface, located between STA 639 and STA 663, on top of wing, aft of overwing exits.
See Doc. D626A001 - DTR, DTR check form 57-10-05-3a, for alternative inspection.

AIRPLANE NOTE: All except 900ER

ACCESS NOTE: Inspection requires removal of wing-to-body fairing.

57-601-10-02	AWL	57-05-02-130-802	1.1	56000 FC	18000 FC	600 700 700C 700IGW 800 900	ALL

Inspect (Ultrasonic) the double plus chord vertical flange at body stringer 18A interface, located between STA 639 and STA 663, on top of wing, aft of overwing exits.
See Doc. D626A001 - DTR, DTR check form 57-10-05-3a, for alternative inspection.

AIRPLANE NOTE: All except 900ER

ACCESS NOTE: Inspection requires removal of wing-to-body fairing.

57-601-22-01	AWL	57-05-02-210-823	1.1	NOTE		ALL	ALL

Inspect (General Visual) the upper horizontal flange at the double plus chord from the front spar to rear spar. Inspection is on both the inboard and outboard locations at BBL 70.85.
See Doc. D626A001 - DTR, DTR check form 57-10-05-4, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

ACCESS NOTE: Inspection requires removal of the wing-to-body fairing and floor panels.

57-601-22-02	AWL	57-05-02-210-823	1.1	NOTE		ALL	ALL

Inspect (General Visual) the upper horizontal flange at the double plus chord from the front spar to rear spar. Inspection is on both the inboard and outboard locations at BBL 70.85.
See Doc. D626A001 - DTR, DTR check form 57-10-05-4, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

ACCESS NOTE: Inspection requires removal of the wing-to-body fairing and floor panels.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-601-25-01	AWL	57-05-02-250-846	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the upper horizontal flange of the chord at the stub beams at STAs 559, 578, 597, 616 and 639.
See Doc. D626A001 - DTR, DTR check form 57-10-05-5, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-43.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

ACCESS NOTE: Floor panel removal is required.

57-601-25-02	AWL	57-05-02-250-846	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the upper horizontal flange of the chord at the stub beams at STAs 559, 578, 597, 616 and 639.
See Doc. D626A001 - DTR, DTR check form 57-10-05-5, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-43.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

ACCESS NOTE: Floor panel removal is required.

57-601-27-01	AWL	57-05-02-210-824	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (General Visual) the upper vertical flange at the double plus chord forward of STA 639.
See Doc D626A001-DTR, DTR Check Form 57-10-05-6 For alternative inspections.

ACCESS NOTE: Inspection requires removal of wing-to-body fairing.

57-601-27-02	AWL	57-05-02-210-824	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (General Visual) the upper vertical flange at the double plus chord forward of STA 639.
See Doc D626A001-DTR, DTR Check Form 57-10-05-6 For alternative inspections.

ACCESS NOTE: Inspection requires removal of wing-to-body fairing.

57-601-28-01	AWL	57-05-02-250-921	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the upper vertical flange at the double plus chord forward of STA 639.
See Doc. D626A001 - DTR, DTR check form 57-10-05-6, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-10-77.

ACCESS NOTE: Inspection requires removal of wing-to-body fairing.

57-601-28-02	AWL	57-05-02-250-921	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the upper vertical flange at the double plus chord forward of STA 639.
See Doc. D626A001 - DTR, DTR check form 57-10-05-6, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-10-77.

ACCESS NOTE: Inspection requires removal of wing-to-body fairing.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-601-30-01	AWL	57-05-02-130-804	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) the wing center section upper skin at floor beams and shear ties located at BL 0, BL 25, BL 45, from front spar to rear spar.
See Doc. D626A001 - DTR, DTR check form 57-10-06, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-12.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.

ACCESS NOTE: Floor panel removal is required.

57-601-30-02	AWL	57-05-02-130-804	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) the wing center section upper skin at floor beams and shear ties located at BL 0, BL 25, BL 45, from front spar to rear spar.
See Doc. D626A001 - DTR, DTR check form 57-10-06, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-12.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.

ACCESS NOTE: Floor panel removal is required.

57-602-00-01	AWL	57-05-02-211-802	1.1	NOTE		ALL	ALL

Inspect (Detailed) the wing center section lower panel stringers at stringer No.1 through No. 4 and stringer No. 6 through No. 8 from LBL 67.0 to RBL 67.0.
See Doc. D626A001 - DTR, DTR check form 57-10-07-1, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

57-602-00-02	AWL	57-05-02-211-802	1.1	NOTE		ALL	ALL

Inspect (Detailed) the wing center section lower panel stringers at stringer No.1 through No. 4 and stringer No. 6 through No. 8 from LBL 67.0 to RBL 67.0.
See Doc. D626A001 - DTR, DTR check form 57-10-07-1, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

57-602-10-01	AWL	57-05-02-211-803	1.1	NOTE		ALL	ALL

Inspect (Detailed) wing center section, lower panel skin from the front spar to rear spar and side of body to side of body.
See Doc. D626A001 - DTR, DTR check form 57-10-07/08, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

57-602-10-02	AWL	57-05-02-211-803	1.1	NOTE		ALL	ALL

Inspect (Detailed) wing center section, lower panel skin from the front spar to rear spar and side of body to side of body.
See Doc. D626A001 - DTR, DTR check form 57-10-07/08, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-603-00-01	AWL	57-05-02-211-804	1.1	NOTE		ALL	ALL

Inspect (Detailed) wing center section lower splice stringer No. 5 and No. 9 from LBL 67.0 to RBL 67.0.
See Doc. D626A001 - DTR, DTR check form 57-10-08, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 15000FC.

57-603-00-02	AWL	57-05-02-211-804	1.1	NOTE		ALL	ALL

Inspect (Detailed) wing center section lower splice stringer No. 5 and No. 9 from LBL 67.0 to RBL 67.0.
See Doc. D626A001 - DTR, DTR check form 57-10-08, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 15000FC.

57-604-00-01	AWL	57-05-02-210-801	1.1	NOTE		ALL	ALL

Inspect (General Visual) the wing center section lower panel at the rear spar chord from LBL 67.0 to RBL 67.0.
See Doc. D626A001 - DTR, DTR check form 57-10-09, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

57-604-00-02	AWL	57-05-02-210-801	1.1	NOTE		ALL	ALL

Inspect (General Visual) the wing center section lower panel at the rear spar chord from LBL 67.0 to RBL 67.0.
See Doc. D626A001 - DTR, DTR check form 57-10-09, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

57-605-00-01	AWL	57-05-02-250-802	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the side of body splice lower surface at the lower tee chord from the front spar to the rear spar at BBL 70.85. Inspection is on both the inboard and outboard locations at BBL 70.85.
See Doc. D626A001 - DTR, DTR check form 57-10-11-1, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-62.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

ACCESS NOTE: Inspection requires removal of Wing-to-Body fairing.

57-605-00-02	AWL	57-05-02-250-802	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the side of body splice lower surface at the lower tee chord from the front spar to the rear spar at BBL 70.85. Inspection is on both the inboard and outboard locations at BBL 70.85.
See Doc. D626A001 - DTR, DTR check form 57-10-11-1, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-62.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

ACCESS NOTE: Inspection requires removal of Wing-to-Body fairing.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-605-01-01	AWL	57-05-02-250-803	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the side of body splice lower surface at the lower tee chord from the front spar to the rear spar at BBL 70.85.

See Doc. D626A001 - DTR, DTR check form 57-10-11-1, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-35.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-605-01-02	AWL	57-05-02-250-803	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the side of body splice lower surface at the lower tee chord from the front spar to the rear spar at BBL 70.85.

See Doc. D626A001 - DTR, DTR check form 57-10-11-1, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-35.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-605-10-01	AWL	57-05-02-250-850	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the side of body splice, lower surface skin in the non hidden areas from the front spar to the rear spar at BBL 70.85.

See Doc. D626A001 - DTR, DTR check form 57-10-11-2, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-39.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

ACCESS NOTE: Inspection requires removal of Wing-to-Body fairing.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-605-10-02	AWL	57-05-02-250-850	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the side of body splice, lower surface skin in the non hidden areas from the front spar to the rear spar at BBL 70.85.

See Doc. D626A001 - DTR, DTR check form 57-10-11-2, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-39.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

ACCESS NOTE: Inspection requires removal of Wing-to-Body fairing.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-605-20-01	AWL	57-05-02-250-852	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the side of body splice, lower surface skin in the hidden areas from the front spar to the rear spar at BBL 70.85.

See Doc. D626A001 - DTR, DTR check form 57-10-11-3, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-70.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

ACCESS NOTE: Inspection requires removal of Wing-to-Body fairing.

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-605-20-02	AWL	57-05-02-250-852	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the side of body splice, lower surface skin in the hidden areas from the front spar to the rear spar at BBL 70.85.

See Doc. D626A001 - DTR, DTR check form 57-10-11-3, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-70.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

ACCESS NOTE: Inspection requires removal of Wing-to-Body fairing.

57-606-00-01	AWL	57-05-02-130-805	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) the lower panel at attachment to the keel beam from rear spar to the front spar.

See Doc. D626A001 - DTR, DTR check form 57-10-12, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-13.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 9000FC.

57-606-00-02	AWL	57-05-02-130-805	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) the lower panel at attachment to the keel beam from rear spar to the front spar.

See Doc. D626A001 - DTR, DTR check form 57-10-12, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-13.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 9000FC.

57-606-01-01	AWL	57-05-02-250-806	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the lower panel skin at the drain installation between stringer S-7 and stringer S-8 at LBBL 3.5.

See Doc. D626A001 - DTR, DTR check form 57-10-13, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-75.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

57-606-01-02	AWL	57-05-02-250-806	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the lower panel skin at the drain installation between stringer S-7 and stringer S-8 at LBBL 3.5.

See Doc. D626A001 - DTR, DTR check form 57-10-13, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-75.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-607-00-01	AWL	57-05-02-130-806	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) the lower skin at the lower beam attachment from the front spar to the rear spar.
See Doc. D626A001 - DTR, DTR check form 57-10-15, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-13.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 9000FC.

57-607-00-02	AWL	57-05-02-130-806	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) the lower skin at the lower beam attachment from the front spar to the rear spar.
See Doc. D626A001 - DTR, DTR check form 57-10-15, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-13.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 9000FC.

57-607-10-01	AWL	57-05-02-211-805	1.1	NOTE		ALL	ALL

Inspect (Detailed) the web common to the fuel tank from LBBL 70.85 to RBBL 70.85.
See Doc. D626A001 - DTR, DTR check form 57-10-17-1, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 4000FC.

57-607-10-02	AWL	57-05-02-211-805	1.1	NOTE		ALL	ALL

Inspect (Detailed) the web common to the fuel tank from LBBL 70.85 to RBBL 70.85.
See Doc. D626A001 - DTR, DTR check form 57-10-17-1, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 4000FC.

57-607-11-01	AWL	57-05-02-250-807	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the web common to the fuel tank from LBBL 70.85 to RBBL 70.85.
See Doc. D626A001 - DTR, DTR check form 57-10-17-1, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-81.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.

57-607-11-02	AWL	57-05-02-250-807	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the web common to the fuel tank from LBBL 70.85 to RBBL 70.85.
See Doc. D626A001 - DTR, DTR check form 57-10-17-1, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-81.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-607-20-01	AWL	57-05-02-211-806	1.1	NOTE		ALL	ALL
	Inspect (Detailed) the web common to the fuel tank from LBBL 70.85 to RBBL 70.85. See Doc. D626A001 - DTR, DTR check form 57-10-17-2, for alternative inspection. **INTERVAL NOTE:** Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 8000FC.						
57-607-20-02	AWL	57-05-02-211-806	1.1	NOTE		ALL	ALL
	Inspect (Detailed) the web common to the fuel tank from LBBL 70.85 to RBBL 70.85. See Doc. D626A001 - DTR, DTR check form 57-10-17-2, for alternative inspection. **INTERVAL NOTE:** Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 8000FC.						
57-607-21-01	AWL	57-05-02-250-922	1.1	NOTE		ALL	ALL
	Inspect (Low Frequency Eddy Current) the web common to the fuel tank from LBBL 70.85 to RBBL 70.85. See Doc. D626A001 - DTR, DTR check form 57-10-17-2, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-81. **INTERVAL NOTE:** Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.						
57-607-21-02	AWL	57-05-02-250-922	1.1	NOTE		ALL	ALL
	Inspect (Low Frequency Eddy Current) the web common to the fuel tank from LBBL 70.85 to RBBL 70.85. See Doc. D626A001 - DTR, DTR check form 57-10-17-2, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-81. **INTERVAL NOTE:** Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.						
57-607-30-01	AWL	57-05-02-250-808	1.1	56000 FC	22600 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) rear spar keel beam stiffeners at LBL 6.2 and RBL 6.2. See Doc. D626A001 - DTR, DTR check form 57-10-18, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-10-74.						
57-607-30-02	AWL	57-05-02-250-808	1.1	56000 FC	22600 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) rear spar keel beam stiffeners at LBL 6.2 and RBL 6.2. See Doc. D626A001 - DTR, DTR check form 57-10-18, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-10-74.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-608-00-01	AWL	57-05-02-250-809	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) around the four AFT fasteners along the rear spar that attach the splice plate to the wing lower skin and center section lower skin at BBL 70.85 as well as the upper rear spar through the double plus chord horizontal flange on both sides of the joint.

See Doc. D626A001 - DTR, DTR check form 57-10-20, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-17.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

ACCESS NOTE: Inspection requires removal of external panels.

57-608-00-02	AWL	57-05-02-250-809	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) around the four AFT fasteners along the rear spar that attach the splice plate to the wing lower skin and center section lower skin at BBL 70.85 as well as the upper rear spar through the double plus chord horizontal flange on both sides of the joint.

See Doc. D626A001 - DTR, DTR check form 57-10-20, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-17.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

ACCESS NOTE: Inspection requires removal of external panels.

57-609-00-01	AWL	57-05-02-250-811	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) around the four FWD fasteners along the front spar that attach the splice plate to the wing lower skin and center section lower skin at BBL 70.85 as well as the upper front spar through the double plus chord horizonal flange on both sides of the joint.

See Doc. D626A001 - DTR, DTR check form 57-10-21, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-17.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

ACCESS NOTE: Inspection requires removal of the external fairings.

57-609-00-02	AWL	57-05-02-250-811	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) around the four FWD fasteners along the front spar that attach the splice plate to the wing lower skin and center section lower skin at BBL 70.85 as well as the upper front spar through the double plus chord horizonal flange on both sides of the joint.

See Doc. D626A001 - DTR, DTR check form 57-10-21, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-17.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

ACCESS NOTE: Inspection requires removal of the external fairings.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-609-10-01	AWL	57-05-02-250-813	1.1	56000 FC	11000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the BL 0 and BL 25 floor beams from BS 655 to BS 675, and BL 0 floor beam from BS 716 to BS 727B, and BL 45 floor beam from BS 685 to BS 716.

See Doc. D626A001 - DTR, DTR check form 57-10-23-1, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-10-83.

ACCESS NOTE: Remove passenger cabin floor panels as required.

57-609-10-02	AWL	57-05-02-250-813	1.1	56000 FC	11000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the BL 0 and BL 25 floor beams from BS 655 to BS 675, and BL 0 floor beam from BS 716 to BS 727B, and BL 45 floor beam from BS 685 to BS 716.

See Doc. D626A001 - DTR, DTR check form 57-10-23-1, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-10-83.

ACCESS NOTE: Remove passenger cabin floor panels as required.

57-610-00-01	AWL	57-05-02-211-807	1.1	56000 FC	24000 FC	ALL	ALL

Inspect (Detailed) the BL0, 25, and 45 floor beams from STA 569 to STA 655 and the BL 25 floor beams from STA 685 to STA 727A.

See Doc. D626A001 - DTR, DTR check form 57-10-23-2, for alternative inspection.

ACCESS NOTE: Remove passenger cabin floor panels as required.

57-610-00-02	AWL	57-05-02-211-807	1.1	56000 FC	24000 FC	ALL	ALL

Inspect (Detailed) the BL0, 25, and 45 floor beams from STA 569 to STA 655 and the BL 25 floor beams from STA 685 to STA 727A.

See Doc. D626A001 - DTR, DTR check form 57-10-23-2, for alternative inspection.

ACCESS NOTE: Remove passenger cabin floor panels as required.

57-610-01-01	AWL	57-05-02-250-814	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the lower chord along the radius at BL0, 25, and 45 floor beams from STA 664 to STA 727B.

See Doc. D626A001 - DTR, DTR check form 57-10-23-3, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-10-83.

ACCESS NOTE: Remove passenger cabin floor panels as required.

57-610-01-02	AWL	57-05-02-250-814	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the lower chord along the radius at BL0, 25, and 45 floor beams from STA 664 to STA 727B.

See Doc. D626A001 - DTR, DTR check form 57-10-23-3, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-10-83.

ACCESS NOTE: Remove passenger cabin floor panels as required.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-611-00-01	AWL	57-05-02-250-815	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the lower chord along the radius at the BL 0, 25, 45 floor beams between STA 574 through STA 664.
See Doc. D626A001 - DTR, DTR check form 57-10-23-4, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-10-83.

ACCESS NOTE: Remove passenger cabin floor panels as required. Sealant present in chord radius must be removed for full inspection credit.

57-611-00-02	AWL	57-05-02-250-815	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the lower chord along the radius at the BL 0, 25, 45 floor beams between STA 574 through STA 664.
See Doc. D626A001 - DTR, DTR check form 57-10-23-4, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-10-83.

ACCESS NOTE: Remove passenger cabin floor panels as required. Sealant present in chord radius must be removed for full inspection credit.

57-611-10-01	AWL	57-05-02-211-808	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (Detailed) the lower chord along the BL 0 25, and 45 floor beams between STA 540 through STA 574.
See Doc. D626A001 - DTR, DTR check form 57-10-23-5, for alternative inspection.

ACCESS NOTE: Remove passenger cabin floor panels as required. Remove any sealant beyond specifications for full inspection credit.

57-611-10-02	AWL	57-05-02-211-808	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (Detailed) the lower chord along the BL 0 25, and 45 floor beams between STA 540 through STA 574.
See Doc. D626A001 - DTR, DTR check form 57-10-23-5, for alternative inspection.

ACCESS NOTE: Remove passenger cabin floor panels as required. Remove any sealant beyond specifications for full inspection credit.

57-611-20-01	AWL	57-05-02-211-809	1.1	NOTE		ALL	ALL

Inspect (Detailed) the typical web to stiffener attach points on the side of body rib.
See Doc. D626A001 - DTR, DTR check form 57-10-25, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-611-20-02	AWL	57-05-02-211-809	1.1	NOTE		ALL	ALL

Inspect (Detailed) the typical web to stiffener attach points on the side of body rib.
See Doc. D626A001 - DTR, DTR check form 57-10-25, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-611-21-01	AWL	57-05-02-211-810	1.1	NOTE		ALL	ALL

Inspect (Detailed) the typical web to stiffener attach points on the side of body rib.
See Doc. D626A001 - DTR, DTR check form 57-10-25, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

ACCESS NOTE: Remove passenger cabin floor panels as required.

57-611-21-02	AWL	57-05-02-211-810	1.1	NOTE		ALL	ALL

Inspect (Detailed) the typical web to stiffener attach points on the side of body rib.
See Doc. D626A001 - DTR, DTR check form 57-10-25, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

ACCESS NOTE: Remove passenger cabin floor panels as required.

57-612-00-01	AWL	57-05-02-210-802	1.1	NOTE		ALL	ALL

Inspect (General Visual) stringers S-2 through S-4, and S-10 through S-13 from rib 1 to rib 10 at the non-hidden, fairing areas.
See Doc. D626A001 - DTR, DTR check form 57-20-01-1, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-612-00-02	AWL	57-05-02-210-802	1.1	NOTE		ALL	ALL

Inspect (General Visual) stringers S-2 through S-4, and S-10 through S-13 from rib 1 to rib 10 at the non-hidden, fairing areas.
See Doc. D626A001 - DTR, DTR check form 57-20-01-1, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-612-01-01	AWL	57-05-02-210-804	1.1	NOTE		ALL	ALL

Inspect (General Visual) stringers S-1 through S-4 and S-10 through S-14 from rib 1 to rib 15 adjacent to spar chords at the non hidden areas.
See Doc. D626A001 - DTR, DTR check form 57-20-01-2, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-612-01-02	AWL	57-05-02-210-804	1.1	NOTE		ALL	ALL

Inspect (General Visual) stringers S-1 through S-4 and S-10 through S-14 from rib 1 to rib 15 adjacent to spar chords at the non hidden areas.
See Doc. D626A001 - DTR, DTR check form 57-20-01-2, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-613-00-01	AWL	57-05-02-250-817	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) stringers S-1 through S-4 and S-10 through S-14 at ribs 5 & 8 at the locations hidden by seal pans & sealant.
See Doc. D626A001 - DTR, DTR check form 57-20-01-3, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-79.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 22500FC.

57-613-00-02	AWL	57-05-02-250-817	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) stringers S-1 through S-4 and S-10 through S-14 at ribs 5 & 8 at the locations hidden by seal pans & sealant.
See Doc. D626A001 - DTR, DTR check form 57-20-01-3, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-79.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 22500FC.

57-614-00-01	AWL	57-05-02-130-807	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) the skin panel at the rub strips from rib 5 to rib 8, rib 9 to rib 11 and rib 13 to 15.
See Doc. D626A001 - DTR, DTR check form 57-20-01, -02, -03, -04, -05, -08-1, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-15.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 2750FC.

57-614-00-02	AWL	57-05-02-130-807	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) the skin panel at the rub strips from rib 5 to rib 8, rib 9 to rib 11 and rib 13 to 15.
See Doc. D626A001 - DTR, DTR check form 57-20-01, -02, -03, -04, -05, -08-1, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-15.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 2750FC.

57-614-10-01	AWL	57-05-02-210-806	1.1	NOTE		ALL	ALL

Inspect (General Visual) rib 1 to rib 14 at the externally visible areas from the front spar to the rear spar.
See Doc. D626A001 - DTR, DTR check form 57-20-01, -02, -03, -04, -05, -08-2, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 4000FC.

57-614-10-02	AWL	57-05-02-210-806	1.1	NOTE		ALL	ALL

Inspect (General Visual) rib 1 to rib 14 at the externally visible areas from the front spar to the rear spar.
See Doc. D626A001 - DTR, DTR check form 57-20-01, -02, -03, -04, -05, -08-2, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 4000FC.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-614-20-01	AWL	57-05-02-211-813	1.1	NOTE		ALL	ALL
	Inspect (Detailed) rib 1 to rib 27 at the skin not covered by rub strips or fittings. See Doc. D626A001 - DTR, DTR check form 57-20-01, -02, -03, -04, -05, -08-3, for alternative inspection. **INTERVAL NOTE:** Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 3000FC. **ACCESS NOTE:** Fairing removal required.						
57-614-20-02	AWL	57-05-02-211-813	1.1	NOTE		ALL	ALL
	Inspect (Detailed) rib 1 to rib 27 at the skin not covered by rub strips or fittings. See Doc. D626A001 - DTR, DTR check form 57-20-01, -02, -03, -04, -05, -08-3, for alternative inspection. **INTERVAL NOTE:** Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 3000FC. **ACCESS NOTE:** Fairing removal required.						
57-614-30-01	AWL	57-05-02-210-808	1.1	NOTE		ALL	ALL
	Inspect (General Visual) rib 14 to rib 27 at the externally visible areas from the front spar to the rear spar. See Doc. D626A001 - DTR, DTR check form 57-20-01, -02, -03, -04, -05, -08-4, for alternative inspection. **INTERVAL NOTE:** Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 5000FC.						
57-614-30-02	AWL	57-05-02-210-808	1.1	NOTE		ALL	ALL
	Inspect (General Visual) rib 14 to rib 27 at the externally visible areas from the front spar to the rear spar. See Doc. D626A001 - DTR, DTR check form 57-20-01, -02, -03, -04, -05, -08-4, for alternative inspection. **INTERVAL NOTE:** Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 5000FC.						
57-614-40-01	AWL	57-05-02-211-815	1.1	NOTE		ALL	ALL
	Inspect (Detailed) stringer S-6 and S-8 from rib 1 to rib 19 at the non hidden, faired and non-faired areas. See Doc. D626A001 - DTR, DTR check form 57-20-02-1, for alternative inspection. **INTERVAL NOTE:** Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.						
57-614-40-02	AWL	57-05-02-211-815	1.1	NOTE		ALL	ALL
	Inspect (Detailed) stringer S-6 and S-8 from rib 1 to rib 19 at the non hidden, faired and non-faired areas. See Doc. D626A001 - DTR, DTR check form 57-20-02-1, for alternative inspection. **INTERVAL NOTE:** Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-615-00-01	AWL	57-05-02-130-809	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) the vertical flange and horizontal attachment flange on stringer S-6 and S-8, ribs 5 and 8, where stringers are hidden under seal pan and sealant.
See Doc. D626A001 - DTR, DTR check form 57-20-02-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-07.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

57-615-00-02	AWL	57-05-02-130-809	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) the vertical flange and horizontal attachment flange on stringer S-6 and S-8, ribs 5 and 8, where stringers are hidden under seal pan and sealant.
See Doc. D626A001 - DTR, DTR check form 57-20-02-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-07.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

57-615-01-01	AWL	57-05-02-250-855	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the lower wing panel rail stringer, S-8, from rib 6 to rib 7 at the areas hidden by the nacelle support strut attachment fitting.
See Doc. D626A001 - DTR, DTR check form 57-20-02-3, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-80.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.

57-615-01-02	AWL	57-05-02-250-855	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the lower wing panel rail stringer, S-8, from rib 6 to rib 7 at the areas hidden by the nacelle support strut attachment fitting.
See Doc. D626A001 - DTR, DTR check form 57-20-02-3, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-80.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.

57-616-00-01	AWL	57-05-02-210-810	1.1	NOTE		ALL	ALL

Inspect (General Visual) the lower wing panel rail stringers, S-6 and S-8, from rib 19 to rib 25 at the non-hidden areas.
See Doc. D626A001 - DTR, DTR check form 57-20-02-4, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-616-00-02	AWL	57-05-02-210-810	1.1	NOTE		ALL	ALL

Inspect (General Visual) the lower wing panel rail stringers, S-6 and S-8, from rib 19 to rib 25 at the non-hidden areas.
See Doc. D626A001 - DTR, DTR check form 57-20-02-4, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-616-10-01	AWL	57-05-02-130-811	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) lower wing panel rail stringers, S-6 and S-8, at the areas hidden by the flange of rib 22, shims and sealant.

See Doc. D626A001 - DTR, DTR check form 57-20-02-5, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-07.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-616-10-02	AWL	57-05-02-130-811	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) lower wing panel rail stringers, S-6 and S-8, at the areas hidden by the flange of rib 22, shims and sealant.

See Doc. D626A001 - DTR, DTR check form 57-20-02-5, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-07.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-617-00-01	AWL	57-05-02-210-812	1.1	NOTE		ALL	ALL

Inspect (General Visual) the lower wing panel splice stringers, S-5 and S-9, from rib 1 to rib 10 at the non-hidden areas without the fairing.

See Doc. D626A001 - DTR, DTR check form 57-20-03-1, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-617-00-02	AWL	57-05-02-210-812	1.1	NOTE		ALL	ALL

Inspect (General Visual) the lower wing panel splice stringers, S-5 and S-9, from rib 1 to rib 10 at the non-hidden areas without the fairing.

See Doc. D626A001 - DTR, DTR check form 57-20-03-1, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-617-01-01	AWL	57-05-02-250-819	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the lower wing panel splice stringers, S-5 and S-9, from rib 1 to rib 10 at the non-hidden areas without the fairing.

See Doc. D626A001 - DTR, DTR check form 57-20-03-1, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-55.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 5500FC.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-617-01-02	AWL	57-05-02-250-819	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the lower wing panel splice stringers, S-5 and S-9, from rib 1 to rib 10 at the non-hidden areas without the fairing.
See Doc. D626A001 - DTR, DTR check form 57-20-03-1, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-55.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 5500FC.

57-618-00-01	AWL	57-05-02-211-875	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the lower wing panel splice stringers, S-5 and S-9, from rib 1 to rib 10 at the non-hidden areas under the fairing.
See Doc. D626A001 - DTR, DTR check form 57-20-03-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-33.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 15000FC.

57-618-00-02	AWL	57-05-02-211-875	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the lower wing panel splice stringers, S-5 and S-9, from rib 1 to rib 10 at the non-hidden areas under the fairing.
See Doc. D626A001 - DTR, DTR check form 57-20-03-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-33.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 15000FC.

57-618-01-01	AWL	57-05-02-250-917	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the lower wing panel splice stringers, S-5 and S-9, from rib 1 to rib 10 at the non-hidden areas under the fairing.
See Doc. D626A001-DTR, DTR check form 57-20-03-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-33.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 15000FC.

57-618-01-02	AWL	57-05-02-250-917	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the lower wing panel splice stringers, S-5 and S-9, from rib 1 to rib 10 at the non-hidden areas under the fairing.
See Doc. D626A001-DTR, DTR check form 57-20-03-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-33.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 15000FC.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-619-00-01	AWL	57-05-02-130-813	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) the lower wing panel splice stringers, S-5 and S-9, at rib 5 and rib 8 at the areas hidden by seal pans and sealant.
See Doc. D626A001 - DTR, DTR check form 57-20-03-3, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-07.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-619-00-02	AWL	57-05-02-130-813	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) the lower wing panel splice stringers, S-5 and S-9, at rib 5 and rib 8 at the areas hidden by seal pans and sealant.
See Doc. D626A001 - DTR, DTR check form 57-20-03-3, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-07.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-619-01-01	AWL	57-05-02-250-821	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the lower wing panel splice stringers, S-5 and S-9, from rib 5 to rib 8 at the areas hidden by seal pans and sealant.
See Doc. D626A001 - DTR, DTR check form 57-20-03-3, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-55.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 5500FC.

57-619-01-02	AWL	57-05-02-250-821	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the lower wing panel splice stringers, S-5 and S-9, from rib 5 to rib 8 at the areas hidden by seal pans and sealant.
See Doc. D626A001 - DTR, DTR check form 57-20-03-3, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-55.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 5500FC.

57-620-00-01	AWL	57-05-02-211-817	1.1	NOTE		ALL	ALL

Inspect (Detailed) the web and free flange of the lower wing panel splice stringers, S-5 and S-9, from rib 10 to rib 19 except at areas externally covered by rub strips.
See Doc. D626A001 - DTR, DTR check form 57-20-03-4, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 15000FC.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-620-00-02	AWL	57-05-02-211-817	1.1	NOTE		ALL	ALL

Inspect (Detailed) the web and free flange of the lower wing panel splice stringers, S-5 and S-9, from rib 10 to rib 19 except at areas externally covered by rub strips.
See Doc. D626A001 - DTR, DTR check form 57-20-03-4, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 15000FC.

57-620-01-01	AWL	57-05-02-250-825	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the lower wing panel splice stringers, S-5 and S-9, from rib 10 to rib 19, except at areas externally covered by rub strips.
See Doc. D626A001 - DTR, DTR check form 57-20-03-4, for alternative repeat inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-55.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 5500FC.

ACCESS NOTE: Fairing removal required.

57-620-01-02	AWL	57-05-02-250-825	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the lower wing panel splice stringers, S-5 and S-9, from rib 10 to rib 19, except at areas externally covered by rub strips.
See Doc. D626A001 - DTR, DTR check form 57-20-03-4, for alternative repeat inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-55.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 5500FC.

ACCESS NOTE: Fairing removal required.

57-621-00-01	AWL	57-05-02-211-878	1.1	NOTE		ALL	ALL

Inspect (Detailed) the web and free flange of the wing lower panel splice stringers, S-5 and S-9, from rib 10 to rib 19 at the areas externally covered by rub strips.
See Doc. D626A001 - DTR, DTR check form 57-20-03-5, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 15000 FC.

57-621-00-02	AWL	57-05-02-211-878	1.1	NOTE		ALL	ALL

Inspect (Detailed) the web and free flange of the wing lower panel splice stringers, S-5 and S-9, from rib 10 to rib 19 at the areas externally covered by rub strips.
See Doc. D626A001 - DTR, DTR check form 57-20-03-5, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 15000 FC.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-621-01-01	AWL	57-05-02-250-827	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the wing lower panel splice stringers, S-5 and S-9, from rib 10 to rib 19, at the areas externally covered by rub strips.
See Doc. D626A001 - DTR, DTR check form 57-20-03-5, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-33.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 15000 FC.

ACCESS NOTE: Remove minimal amount of sealant to facilitate direction 3 HFEC in the lower stringer radius.

57-621-01-02	AWL	57-05-02-250-827	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the wing lower panel splice stringers, S-5 and S-9, from rib 10 to rib 19, at the areas externally covered by rub strips.
See Doc. D626A001 - DTR, DTR check form 57-20-03-5, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-33.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 15000 FC.

ACCESS NOTE: Remove minimal amount of sealant to facilitate direction 3 HFEC in the lower stringer radius.

57-621-02-01	AWL	57-05-02-250-828	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the wing lower panel splice stringers, S-5 and S-9, from rib 10 to rib 19, at the areas externally covered by rub strips.
See Doc. D626A001 - DTR, DTR check form 57-20-03-5, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-33.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 15000 FC.

ACCESS NOTE: Remove minimal amount of sealant to facilitate direction 4 HFEC at all fasteners.

57-621-02-02	AWL	57-05-02-250-828	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the wing lower panel splice stringers, S-5 and S-9, from rib 10 to rib 19, at the areas externally covered by rub strips.
See Doc. D626A001 - DTR, DTR check form 57-20-03-5, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-33.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 15000 FC.

ACCESS NOTE: Remove minimal amount of sealant to facilitate direction 4 HFEC at all fasteners.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-621-10-01	AWL	57-05-02-250-831	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the lower wing panel splice stringers, S-5 and S-9, from rib 1 to rib 10 at the areas externally covered by fairing and rub strips.
See Doc. D626A001 - DTR, DTR check form 57-20-03-6, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-33.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 15000FC.

ACCESS NOTE: Remove minimal amount of sealant to facilitate direction 3 HFEC in the lower stringer radius.

57-621-10-02	AWL	57-05-02-250-831	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the lower wing panel splice stringers, S-5 and S-9, from rib 1 to rib 10 at the areas externally covered by fairing and rub strips.
See Doc. D626A001 - DTR, DTR check form 57-20-03-6, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-33.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 15000FC.

ACCESS NOTE: Remove minimal amount of sealant to facilitate direction 3 HFEC in the lower stringer radius.

57-621-11-01	AWL	57-05-02-250-924	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the lower wing panel splice stringers, S-5 and S-9, from rib 1 to rib 10 at the areas externally covered by fairing and rub strips.
See Doc. D626A001 - DTR, DTR check form 57-20-03-6, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-33.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 15000FC.

ACCESS NOTE: Remove minimal amount of sealant to facilitate direction 4 HFEC at all fasteners.

57-621-11-02	AWL	57-05-02-250-924	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the lower wing panel splice stringers, S-5 and S-9, from rib 1 to rib 10 at the areas externally covered by fairing and rub strips.
See Doc. D626A001 - DTR, DTR check form 57-20-03-6, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-33.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 15000FC.

ACCESS NOTE: Remove minimal amount of sealant to facilitate direction 4 HFEC at all fasteners.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-621-20-01	AWL	57-05-02-250-833	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the splice stringers at the chordwise skin splices on the lower stringers, S-9 from rib 17 to rib 18; S-5 from rib 18 to rib 19 and upper stringer, S-14 from rib 19 to rib 20.
See Doc. D626A001 - DTR, DTR check form 57-20-03/15, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-60.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 5500FC.

57-621-20-02	AWL	57-05-02-250-833	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the splice stringers at the chordwise skin splices on the lower stringers, S-9 from rib 17 to rib 18; S-5 from rib 18 to rib 19 and upper stringer, S-14 from rib 19 to rib 20.
See Doc. D626A001 - DTR, DTR check form 57-20-03/15, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-60.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 5500FC.

57-622-00-01	AWL	57-05-02-250-835	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the front spar lower chord at the non-hidden areas from rib 1 to rib 5, and rib 19 to Rib 22, and the hidden areas from rib 1 to rib 6, and from rib 19 to rib 22.
See Doc. D626A001 - DTR, DTR check form 57-20-04-1, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-54.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 12000FC.

57-622-00-02	AWL	57-05-02-250-835	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the front spar lower chord at the non-hidden areas from rib 1 to rib 5, and rib 19 to Rib 22, and the hidden areas from rib 1 to rib 6, and from rib 19 to rib 22.
See Doc. D626A001 - DTR, DTR check form 57-20-04-1, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-54.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 12000FC.

57-623-00-01	AWL	57-05-02-211-819	1.1	NOTE		ALL	ALL

Inspect (Detailed) the front spar lower chord at the non-hidden areas from rib 5 to rib 7.
See Doc. D626A001 - DTR, DTR check form 57-20-04-2, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-623-00-02	AWL	57-05-02-211-819	1.1	NOTE		ALL	ALL

Inspect (Detailed) the front spar lower chord at the non-hidden areas from rib 5 to rib 7.
See Doc. D626A001 - DTR, DTR check form 57-20-04-2, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-624-00-01	AWL	57-05-02-130-815	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) the front spar lower chord at the areas hidden by stiffeners, rib posts or fittings from ribs 1 to 19 and ribs 22 to 25.

See Doc. D626A001 - DTR, DTR check form 57-20-04-3, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-09.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-624-00-02	AWL	57-05-02-130-815	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) the front spar lower chord at the areas hidden by stiffeners, rib posts or fittings from ribs 1 to 19 and ribs 22 to 25.

See Doc. D626A001 - DTR, DTR check form 57-20-04-3, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-09.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-625-00-01	AWL	57-05-02-250-837	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the front spar lower chord at the non-hidden areas from rib 7 to rib 19, and from rib 22 to rib 25, and the hidden areas from rib 6 to rib 19, and from rib 22 to rib 25.

See Doc. D626A001 - DTR, DTR check form 57-20-04-4, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-54.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-625-00-02	AWL	57-05-02-250-837	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the front spar lower chord at the non-hidden areas from rib 7 to rib 19, and from rib 22 to rib 25, and the hidden areas from rib 6 to rib 19, and from rib 22 to rib 25.

See Doc. D626A001 - DTR, DTR check form 57-20-04-4, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-54.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-626-00-01	AWL	57-05-02-211-821	1.1	NOTE		ALL	ALL

Inspect (Detailed) the front spar lower chord at the non-hidden areas from rib 1 to rib 19.

See Doc. D626A001 - DTR, DTR check form 57-20-04-5, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.

57-626-00-02	AWL	57-05-02-211-821	1.1	NOTE		ALL	ALL

Inspect (Detailed) the front spar lower chord at the non-hidden areas from rib 1 to rib 19.

See Doc. D626A001 - DTR, DTR check form 57-20-04-5, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-627-00-01	AWL	57-05-02-250-868	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the front spar lower chord at the non-hidden areas from rib 19 to rib 22.
See Doc. D626A001 - DTR, DTR check form 57-20-04-6, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-44.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-627-00-02	AWL	57-05-02-250-868	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the front spar lower chord at the non-hidden areas from rib 19 to rib 22.
See Doc. D626A001 - DTR, DTR check form 57-20-04-6, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-44.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-627-01-01	AWL	57-05-02-250-839	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the front spar lower chord at the non-hidden areas from rib 19 to rib 22.
See Doc. D626A001 - DTR, DTR check form 57-20-04-6, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-54.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-627-01-02	AWL	57-05-02-250-839	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the front spar lower chord at the non-hidden areas from rib 19 to rib 22.
See Doc. D626A001 - DTR, DTR check form 57-20-04-6, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-54.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-628-00-01	AWL	57-05-02-130-817	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) the front spar lower chord areas hidden by stiffeners, ribs, posts or fittings from rib 19 to rib 22.
See Doc. D626A001 - DTR, DTR check form 57-20-04-7, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-09.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-628-00-02	AWL	57-05-02-130-817	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) the front spar lower chord areas hidden by stiffeners, ribs, posts or fittings from rib 19 to rib 22.
See Doc. D626A001 - DTR, DTR check form 57-20-04-7, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-09.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-628-01-01	AWL	57-05-02-250-925	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the front spar lower chord areas hidden by stiffeners, ribs, posts or fittings from rib 19 to rib 22.
See Doc. D626A001 - DTR, DTR check form 57-20-04-7, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-54.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-628-01-02	AWL	57-05-02-250-925	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the front spar lower chord areas hidden by stiffeners, ribs, posts or fittings from rib 19 to rib 22.
See Doc. D626A001 - DTR, DTR check form 57-20-04-7, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-54.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-628-10-01	AWL	57-05-02-210-814	1.1	NOTE		ALL	ALL

Inspect (General Visual) the front spar lower chord at the non-hidden areas from rib 22 to rib 25.
See Doc. D626A001 - DTR, DTR check form 57-20-04-8, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-628-10-02	AWL	57-05-02-210-814	1.1	NOTE		ALL	ALL

Inspect (General Visual) the front spar lower chord at the non-hidden areas from rib 22 to rib 25.
See Doc. D626A001 - DTR, DTR check form 57-20-04-8, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-629-00-01	AWL	57-05-02-130-819	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) all fasteners, outer location, common to the R7/R8 nacelle fitting attachment.
See Doc. D626A001 - DTR, DTR check form 57-20-04-9, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-05.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 7000FC.

ACCESS NOTE: Nacelle fairing should be removed for inspection.

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-629-00-02	AWL	57-05-02-130-819	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) all fasteners, outer location, common to the R7/R8 nacelle fitting attachment.

See Doc. D626A001 - DTR, DTR check form 57-20-04-9, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-05.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 7000FC.

ACCESS NOTE: Nacelle fairing should be removed for inspection.

57-630-00-01	AWL	57-05-02-250-841	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) all fasteners, outer location, common to the R7/R8 nacelle fitting attachment.

See Doc. D626A001 - DTR, DTR check form 57-20-04-9, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-51.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 7000FC.

ACCESS NOTE: Nacelle fairing should be removed for inspection.

57-630-00-02	AWL	57-05-02-250-841	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) all fasteners, outer location, common to the R7/R8 nacelle fitting attachment.

See Doc. D626A001 - DTR, DTR check form 57-20-04-9, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-51.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 7000FC.

ACCESS NOTE: Nacelle fairing should be removed for inspection.

57-631-00-01	AWL	57-05-02-130-820	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) all fasteners, inner location, common to the R7/R8 nacelle fitting attachment.

See Doc. D626A001 - DTR, DTR check form 57-20-04-9, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-05.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

ACCESS NOTE: Nacelle fairing should be removed for inspection.

57-631-00-02	AWL	57-05-02-130-820	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) all fasteners, inner location, common to the R7/R8 nacelle fitting attachment.

See Doc. D626A001 - DTR, DTR check form 57-20-04-9, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-05.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

ACCESS NOTE: Nacelle fairing should be removed for inspection.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-632-00-01	AWL	57-05-02-250-842	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) all fasteners, inner location, common to the R7/R8 nacelle fitting attachment.
See Doc. D626A001 - DTR, DTR check form 57-20-04-9, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-51.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

ACCESS NOTE: Nacelle fairing should be removed for inspection.

57-632-00-02	AWL	57-05-02-250-842	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) all fasteners, inner location, common to the R7/R8 nacelle fitting attachment.
See Doc. D626A001 - DTR, DTR check form 57-20-04-9, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-51.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

ACCESS NOTE: Nacelle fairing should be removed for inspection.

57-632-10-01	AWL	57-05-02-250-872	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the front spar lower chord from rib 25 to rib 27.
See Doc. D626A001 - DTR, DTR check form 57-20-04-10, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-54.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-632-10-02	AWL	57-05-02-250-872	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the front spar lower chord from rib 25 to rib 27.
See Doc. D626A001 - DTR, DTR check form 57-20-04-10, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-54.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-633-00-01	AWL	57-05-02-250-919	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the front spar lower chord from rib 17 to rib 18, the rear spar lower chord from rib 18 to rib 19, and the front spar upper chord from rib 19 to rib 20.
See Doc. D626A001 - DTR, DTR check form 57-20-04 / 05 / 16, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-60.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 2750FC.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-633-00-02	AWL	57-05-02-250-919	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the front spar lower chord from rib 17 to rib 18, the rear spar lower chord from rib 18 to rib 19, and the front spar upper chord from rib 19 to rib 20.

See Doc. D626A001 - DTR, DTR check form 57-20-04 / 05 / 16, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-60.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 2750FC.

57-633-01-01	AWL	57-05-02-130-821	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) the front spar lower chord from rib 17 to rib 18, the rear spar lower chord from rib 18 to rib 19, and the front spar upper chord from rib 19 to rib 20.

See Doc. D626A001 - DTR, DTR check form 57-20-04 / 05 / 16, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-14.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-633-01-02	AWL	57-05-02-130-821	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) the front spar lower chord from rib 17 to rib 18, the rear spar lower chord from rib 18 to rib 19, and the front spar upper chord from rib 19 to rib 20.

See Doc. D626A001 - DTR, DTR check form 57-20-04 / 05 / 16, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-14.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-634-00-01	AWL	57-05-02-210-816	1.1	NOTE		ALL	ALL

Inspect (General Visual) the rear spar lower chord at the non-hidden areas from rib 1 to rib 7.

See Doc. D626A001 - DTR, DTR check form 57-20-05-1, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-634-00-02	AWL	57-05-02-210-816	1.1	NOTE		ALL	ALL

Inspect (General Visual) the rear spar lower chord at the non-hidden areas from rib 1 to rib 7.

See Doc. D626A001 - DTR, DTR check form 57-20-05-1, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-635-00-01	AWL	57-05-02-210-818	1.1	NOTE		ALL	ALL

Inspect (General Visual) rear spar lower chord at the non-hidden areas from rib 1 to rib 14.

See Doc. D626A001 - DTR, DTR check form 57-20-05-2, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-635-00-02	AWL	57-05-02-210-818	1.1	NOTE		ALL	ALL

Inspect (General Visual) rear spar lower chord at the non-hidden areas from rib 1 to rib 14.
See Doc. D626A001 - DTR, DTR check form 57-20-05-2, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-636-00-01	AWL	57-05-02-130-823	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) rear spar lower chord at the areas hidden by a stiffener, rib post, or fitting from rib 1 to rib 25.
See Doc. D626A001 - DTR, DTR check form 57-20-05-3, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-16.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

57-636-00-02	AWL	57-05-02-130-823	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) rear spar lower chord at the areas hidden by a stiffener, rib post, or fitting from rib 1 to rib 25.
See Doc. D626A001 - DTR, DTR check form 57-20-05-3, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-16.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

57-637-00-01	AWL	57-05-02-211-823	1.1	NOTE		ALL	ALL

Inspect (Detailed) the rear spar lower chord at the non-hidden areas from rib 7 to rib 14.
See Doc. D626A001 - DTR, DTR check form 57-20-05-4, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.

57-637-00-02	AWL	57-05-02-211-823	1.1	NOTE		ALL	ALL

Inspect (Detailed) the rear spar lower chord at the non-hidden areas from rib 7 to rib 14.
See Doc. D626A001 - DTR, DTR check form 57-20-05-4, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.

57-638-00-01	AWL	57-05-02-250-845	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the rear spar lower chord areas hidden by a stiffener from rib 1 to rib 22 and the areas hidden by rib post or fitting from rib 1 to rib 25.
See Doc. D626A001 - DTR, DTR check form 57-20-05-5, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-58.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-638-00-02	AWL	57-05-02-250-845	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the rear spar lower chord areas hidden by a stiffener from rib 1 to rib 22 and the areas hidden by rib post or fitting from rib 1 to rib 25.
See Doc. D626A001 - DTR, DTR check form 57-20-05-5, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-58.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

57-639-00-01	AWL	57-05-02-211-825	1.1	NOTE		ALL	ALL

Inspect (Detailed) the rear spar lower chord at the non-hidden areas from rib 14 to rib 22.
See Doc. D626A001 - DTR, DTR check form 57-20-05-6, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-639-00-02	AWL	57-05-02-211-825	1.1	NOTE		ALL	ALL

Inspect (Detailed) the rear spar lower chord at the non-hidden areas from rib 14 to rib 22.
See Doc. D626A001 - DTR, DTR check form 57-20-05-6, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-640-00-01	AWL	57-05-02-211-827	1.1	NOTE		ALL	ALL

Inspect (Detailed) the rear spar lower chord at the non-hidden areas from rib 14 to rib 22.
See Doc. D626A001 - DTR, DTR check form 57-20-05-7, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-640-00-02	AWL	57-05-02-211-827	1.1	NOTE		ALL	ALL

Inspect (Detailed) the rear spar lower chord at the non-hidden areas from rib 14 to rib 22.
See Doc. D626A001 - DTR, DTR check form 57-20-05-7, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-641-00-01	AWL	57-05-02-250-874	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the rear spar lower chord from rib 25 to rib 27.
See Doc. D626A001 - DTR, DTR check form 57-20-05-10, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-58.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-641-00-02	AWL	57-05-02-250-874	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the rear spar lower chord from rib 25 to rib 27.

See Doc. D626A001 - DTR, DTR check form 57-20-05-10, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-58.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.

57-642-00-01	AWL	57-05-02-250-847	1.1	56000 FC	28000 FC	600 700 700C 700IGW 800 900	ALL

Inspect (High Frequency Eddy Current) all the fasteners common to the skin and fittings at WSTA 228.25, and 253.00 on the gear beam outboard support fittings and at WSTA 180 and WSTA 190 on the forward trunnion support fitting.

See Doc. D626A001 - DTR, DTR check form 57-20-06 / 07-1, for alternative repeat inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-30-11.

AIRPLANE NOTE: All except 900ER

ACCESS NOTE: Fairing removal required at WSTA 228.25.

57-642-00-02	AWL	57-05-02-250-847	1.1	56000 FC	28000 FC	600 700 700C 700IGW 800 900	ALL

Inspect (High Frequency Eddy Current) all the fasteners common to the skin and fittings at WSTA 228.25, and 253.00 on the gear beam outboard support fittings and at WSTA 180 and WSTA 190 on the forward trunnion support fitting.

See Doc. D626A001 - DTR, DTR check form 57-20-06 / 07-1, for alternative repeat inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-30-11.

AIRPLANE NOTE: All except 900ER

ACCESS NOTE: Fairing removal required at WSTA 228.25.

57-643-00-01	AWL	57-05-02-130-835	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) the lower wing skin at the R2, R3 and R4 nacelle fitting attachments.

See Doc. D626A001 - DTR, DTR check form 57-20-09, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-06.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 2750FC.

57-643-00-02	AWL	57-05-02-130-835	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) the lower wing skin at the R2, R3 and R4 nacelle fitting attachments.

See Doc. D626A001 - DTR, DTR check form 57-20-09, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-06.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 2750FC.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-643-01-01	AWL	57-05-02-250-849	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the lower wing skin at the R2 and R4 nacelle fitting attachments.

See Doc. D626A001 - DTR, DTR check form 57-20-09, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-30-08.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 2750FC.

57-643-01-02	AWL	57-05-02-250-849	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the lower wing skin at the R2 and R4 nacelle fitting attachments.

See Doc. D626A001 - DTR, DTR check form 57-20-09, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-30-08.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 2750FC.

57-643-10-01	AWL	57-05-02-250-851	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the lower wing panel skin at the shear tied rib attachments at rib 14.

See Doc. D626A001 - DTR, DTR check form 57-20-10 for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-64.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

57-643-10-02	AWL	57-05-02-250-851	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the lower wing panel skin at the shear tied rib attachments at rib 14.

See Doc. D626A001 - DTR, DTR check form 57-20-10 for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-64.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

57-644-00-01	AWL	57-05-02-250-853	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the wing lower skin, area under the flap fairing, between the forward and aft attach fittings at flap tracks 1, 2, 7 and 8.

See Doc. D626A001 - DTR, DTR check form 57-20-12, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-30-09.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 12000FC.

ACCESS NOTE: Removal of flap track fairing required.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-644-00-02	AWL	57-05-02-250-853	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the wing lower skin, area under the flap fairing, between the forward and aft attach fittings at flap tracks 1, 2, 7 and 8.
See Doc. D626A001 - DTR, DTR check form 57-20-12, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-30-09.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 12000FC.

ACCESS NOTE: Removal of flap track fairing required.

57-644-01-01	AWL	57-05-02-130-825	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) the perimeter of the forward flap track fitting and aft flap attachment at flap tracks 1, 2, 7 and 8.
See Doc. D626A001 - DTR, DTR check form 57-20-12, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-08.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 12000FC.

ACCESS NOTE: Removal of flap track fairing required.

57-644-01-02	AWL	57-05-02-130-825	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) the perimeter of the forward flap track fitting and aft flap attachment at flap tracks 1, 2, 7 and 8.
See Doc. D626A001 - DTR, DTR check form 57-20-12, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-08.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 12000FC.

ACCESS NOTE: Removal of flap track fairing required.

57-644-02-01	AWL	57-05-02-250-854	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the wing lower skin forward of the flap track attach fittings and between the fairing rub strips at flap tracks 1, 2, 7 and 8.
See Doc. D626A001 - DTR, DTR check form 57-20-12, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-30-09.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 12000FC.

ACCESS NOTE: Removal of flap track fairing required.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-644-02-02	AWL	57-05-02-250-854	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the wing lower skin forward of the flap track attach fittings and between the fairing rub strips at flap tracks 1, 2, 7 and 8.

See Doc. D626A001 - DTR, DTR check form 57-20-12, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-30-09.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 12000FC.

ACCESS NOTE: Removal of flap track fairing required.

57-645-00-01	AWL	57-05-02-211-829	1.1	NOTE		ALL	ALL

Inspect (Detailed) the typical stringers at rib 5 that are hidden under seal pans and sealant.

See Doc. D626A001 - DTR, DTR check form 57-20-13-1, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 6875 FC.

57-645-00-02	AWL	57-05-02-211-829	1.1	NOTE		ALL	ALL

Inspect (Detailed) the typical stringers at rib 5 that are hidden under seal pans and sealant.

See Doc. D626A001 - DTR, DTR check form 57-20-13-1, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 6875 FC.

57-645-05-01	AWL	57-05-02-250-923	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the upper skin-to-plus chord attachment at BBL 70.85. Inspection is on both the inboard and outboard locations at BBL 70.85.

See Doc. D626A001 - DTR, DTR check form 57-20-13/-14/-15/-16/-17-1, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-34.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.

57-645-05-02	AWL	57-05-02-250-923	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the upper skin-to-plus chord attachment at BBL 70.85. Inspection is on both the inboard and outboard locations at BBL 70.85.

See Doc. D626A001 - DTR, DTR check form 57-20-13/-14/-15/-16/-17-1, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-34.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-645-10-01	AWL	57-05-02-250-876	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the forward skin-to-plus chord attachments at splice stringer S-14, BBL 70.85. Inspection is on both the inboard and outboard locations at BBL 70.85.

See Doc. D626A001 - DTR, DTR check form 57-20-13/-14/-15/-16/-17-2, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-34.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.

57-645-10-02	AWL	57-05-02-250-876	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the forward skin-to-plus chord attachments at splice stringer S-14, BBL 70.85. Inspection is on both the inboard and outboard locations at BBL 70.85.

See Doc. D626A001 - DTR, DTR check form 57-20-13/-14/-15/-16/-17-2, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-34.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.

57-645-11-01	AWL	57-05-02-250-878	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the aft skin-to-plus chord attachments at splice stringer S-14, BBL 70.85.

See Doc. D626A001 - DTR, DTR check form 57-20-13/-14/-15/-16/-17-2, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-34.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.

57-645-11-02	AWL	57-05-02-250-878	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the aft skin-to-plus chord attachments at splice stringer S-14, BBL 70.85.

See Doc. D626A001 - DTR, DTR check form 57-20-13/-14/-15/-16/-17-2, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-34.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.

57-645-12-01	AWL	57-05-02-250-880	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the forward skin-to-plus chord attachments at splice stringer S-14, BBL 70.85.

See Doc. D626A001 - DTR, DTR check form 57-20-13/-14/-15/-16/-17-2, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-53.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-645-12-02	AWL	57-05-02-250-880	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the forward skin-to-plus chord attachments at splice stringer S-14, BBL 70.85.
See Doc. D626A001 - DTR, DTR check form 57-20-13/-14/-15/-16/-17-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-53.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.

57-645-13-01	AWL	57-05-02-250-882	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the aft skin-to-plus chord attachments at splice stringer S-14, BBL 70.85.
See Doc. D626A001 - DTR, DTR check form 57-20-13/-14/-15/-16/-17-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-53.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.

57-645-13-02	AWL	57-05-02-250-882	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the aft skin-to-plus chord attachments at splice stringer S-14, BBL 70.85.
See Doc. D626A001 - DTR, DTR check form 57-20-13/-14/-15/-16/-17-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-53.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.

57-645-15-01	AWL	57-05-02-210-821	1.1	NOTE		ALL	ALL

Inspect (General Visual) upper wing panel splice stringer, S-14, from rib 12 to rib 21 at locations adjacent to the spar chord.
See Doc. D626A001 - DTR, DTR check form 57-20-15, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-645-15-02	AWL	57-05-02-210-821	1.1	NOTE		ALL	ALL

Inspect (General Visual) upper wing panel splice stringer, S-14, from rib 12 to rib 21 at locations adjacent to the spar chord.
See Doc. D626A001 - DTR, DTR check form 57-20-15, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-646-00-01	AWL	57-05-02-211-831	1.1	NOTE		ALL	ALL

Inspect (Detailed) rear spar upper chord at the non-hidden areas from rib 1 to rib 13.
See Doc. D626A001 - DTR, DTR check form 57-20-17-1, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-646-00-02	AWL	57-05-02-211-831	1.1	NOTE		ALL	ALL

Inspect (Detailed) rear spar upper chord at the non-hidden areas from rib 1 to rib 13.
See Doc. D626A001 - DTR, DTR check form 57-20-17-1, for alternative inspection.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-647-00-01	AWL	57-05-02-250-857	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the rear spar upper chord at the areas hidden by a stiffener, rib post, or fitting from rib 1 to rib 13.
See Doc. D626A001 - DTR, DTR check form 57-20-17-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-59.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

57-647-00-02	AWL	57-05-02-250-857	1.1	NOTE		ALL	ALL

Inspect (Low Frequency Eddy Current) the rear spar upper chord at the areas hidden by a stiffener, rib post, or fitting from rib 1 to rib 13.
See Doc. D626A001 - DTR, DTR check form 57-20-17-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-59.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

57-647-10-01	AWL	57-05-02-250-859	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the rear spar upper chord at the non-hidden areas from rib 1 to rib 13.
See Doc. D626A001 - DTR, DTR check form 57-20-17-3, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-42.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-647-10-02	AWL	57-05-02-250-859	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the rear spar upper chord at the non-hidden areas from rib 1 to rib 13.
See Doc. D626A001 - DTR, DTR check form 57-20-17-3, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-42.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-647-20-01	AWL	57-05-02-130-827	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) the rear spar upper chord at the areas hidden by stiffener, rib post, or fitting from rib 1 to rib 13.
See Doc. D626A001 - DTR, DTR check form 57-20-17-4, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-10.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-647-20-02	AWL	57-05-02-130-827	1.1	NOTE		ALL	ALL
	Inspect (Ultrasonic) the rear spar upper chord at the areas hidden by stiffener, rib post, or fitting from rib 1 to rib 13. See Doc. D626A001 - DTR, DTR check form 57-20-17-4, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-10. **INTERVAL NOTE:** Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.						
57-648-00-01	AWL	57-05-02-250-861	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the fasteners common to skin and R1 fitting as well as the fasteners common to the fairing bracket from the adjacent edge. See Doc. D626A001 - DTR, DTR check form 57-20-19, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-30-10.						
57-648-00-02	AWL	57-05-02-250-861	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the fasteners common to skin and R1 fitting as well as the fasteners common to the fairing bracket from the adjacent edge. See Doc. D626A001 - DTR, DTR check form 57-20-19, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-30-10.						
57-649-00-01	AWL	57-05-02-211-833	1.1	NOTE		ALL	ALL
	Inspect (Detailed) the web at the stiffeners or fittings from rib 1 to rib 6 and from rib 7 to rib 27. See Doc. D626A001 - DTR, DTR check form 57-20-22, for alternative inspection. **INTERVAL NOTE:** Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.						
57-649-00-02	AWL	57-05-02-211-833	1.1	NOTE		ALL	ALL
	Inspect (Detailed) the web at the stiffeners or fittings from rib 1 to rib 6 and from rib 7 to rib 27. See Doc. D626A001 - DTR, DTR check form 57-20-22, for alternative inspection. **INTERVAL NOTE:** Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.						
57-650-00-01	AWL	57-05-02-211-835	1.1	NOTE		ALL	ALL
	Inspect (Detailed) the outboard wing rear spar web at ribs 1 to 27. See Doc. D626A001 - DTR, DTR check form 57-20-24-1, for alternative inspection. **INTERVAL NOTE:** Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 8000FC.						
57-650-00-02	AWL	57-05-02-211-835	1.1	NOTE		ALL	ALL
	Inspect (Detailed) the outboard wing rear spar web at ribs 1 to 27. See Doc. D626A001 - DTR, DTR check form 57-20-24-1, for alternative inspection. **INTERVAL NOTE:** Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 8000FC.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-650-10-01	AWL	57-05-02-250-884	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the outboard wing rear spar at the hidden areas from rib 1 to rib 27, except at locations covered by PSE 57-20-24/25/26.

See Doc. D626A001 - DTR, DTR check form 57-20-24-2, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-52.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 12000FC.

57-650-10-02	AWL	57-05-02-250-884	1.1	NOTE		ALL	ALL

Inspect (High Frequency Eddy Current) the outboard wing rear spar at the hidden areas from rib 1 to rib 27, except at locations covered by PSE 57-20-24/25/26.

See Doc. D626A001 - DTR, DTR check form 57-20-24-2, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-52.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 12000FC.

57-651-00-01	AWL	57-05-02-130-829	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) the rear spar web at the trunnion attachment and main landing gear fitting locations from rib 1 to rib 27.

See Doc. D626A001 - DTR, DTR check form 57-20-24/25/26, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-11.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

ACCESS NOTE: Deploy flaps and spoilers to gain access.

57-651-00-02	AWL	57-05-02-130-829	1.1	NOTE		ALL	ALL

Inspect (Ultrasonic) the rear spar web at the trunnion attachment and main landing gear fitting locations from rib 1 to rib 27.

See Doc. D626A001 - DTR, DTR check form 57-20-24/25/26, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 4, Subject 57-10-11.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 18000FC.

ACCESS NOTE: Deploy flaps and spoilers to gain access.

57-651-10-01	AWL	57-05-02-250-863	1.1	NOTE		700 800 900ER	ALL

Inspect (High Frequency Eddy Current) the entire upper and lower chord horizontal flange from the front spar to the rear spar at WBL 616.75.

See Doc. D626A001 - DTR, DTR check form 57-20-29-1, for alternative inspection.

The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-40.

AIRPLANE NOTE: Applies to 737-700, 737-800 and 737-900 ER with production installed winglets per Section 9, Table 9-1.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	REPEAT	APPLICABILITY AIRPLANE	ENGINE
57-651-10-02	AWL	57-05-02-250-863	1.1	NOTE		700 800 900ER	ALL

Inspect (High Frequency Eddy Current) the entire upper and lower chord horizontal flange from the front spar to the rear spar at WBL 616.75.
See Doc. D626A001 - DTR, DTR check form 57-20-29-1, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-40.

AIRPLANE NOTE: Applies to 737-700, 737-800 and 737-900 ER with production installed winglets per Section 9, Table 9-1.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 2 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

57-652-00-01	AWL	57-05-02-250-865	1.1	NOTE		700 800 900ER	ALL

Inspect (Low Frequency Eddy Current) the entire upper and lower rib flange to skin attachment from the front spar to the rear spar at WBL 658.17.
See Doc. D626A001 - DTR, DTR check form 57-20-29-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-48.

AIRPLANE NOTE: Applies to 737-700, 737-800 and 737-900 ER with production installed winglets per Section 9, Table 9-1.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.

57-652-00-02	AWL	57-05-02-250-865	1.1	NOTE		700 800 900ER	ALL

Inspect (Low Frequency Eddy Current) the entire upper and lower rib flange to skin attachment from the front spar to the rear spar at WBL 658.17.
See Doc. D626A001 - DTR, DTR check form 57-20-29-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-48.

AIRPLANE NOTE: Applies to 737-700, 737-800 and 737-900 ER with production installed winglets per Section 9, Table 9-1.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 36000FC.

57-653-00-01	AWL	57-05-02-250-867	1.1	NOTE		700 800 900ER	ALL

Inspect (High Frequency Eddy Current) the fasteners holes on the front and rear spar tension fitting at all eight fastener locations from the outboard side passing through rib 27.
See Doc. D626A001 - DTR, DTR check form 57-20-29-3, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-92.

AIRPLANE NOTE: Applies to 737-700, 737-800 and 737-900 ER with production installed winglets per Section 9, Table 9-1.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

ACCESS NOTE: Fastener removal required.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-653-00-02	AWL	57-05-02-250-867	1.1	NOTE		700 800 900ER	ALL

Inspect (High Frequency Eddy Current) the fasteners holes on the front and rear spar tension fitting at all eight fastener locations from the outboard side passing through rib 27.
See Doc. D626A001 - DTR, DTR check form 57-20-29-3, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-92.

AIRPLANE NOTE: Applies to 737-700, 737-800 and 737-900 ER with production installed winglets per Section 9, Table 9-1.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 24000FC.

ACCESS NOTE: Fastener removal required.

57-654-00-01	AWL	57-05-02-211-839	1.1	56000 FC	5000 FC	ALL	ALL

Inspect (Detailed) the links, fittings including the lugs, clevises, and pins.
See Doc. D626A001 - DTR, DTR check form 57-20-35, 57-20-36, 57-20-37, 57-20-38, 57-20-39, for alternative inspection.

57-654-00-02	AWL	57-05-02-211-839	1.1	56000 FC	5000 FC	ALL	ALL

Inspect (Detailed) the links, fittings including the lugs, clevises, and pins.
See Doc. D626A001 - DTR, DTR check form 57-20-35, 57-20-36, 57-20-37, 57-20-38, 57-20-39, for alternative inspection.

57-655-00-01	AWL	57-05-02-250-869	1.1	NOTE		800 900ER	ALL

Inspect (Open Hole Eddy Current) all eighteen upper and lower tension bolt holes common to the winglet STA 0 Rib.
See Doc. D626A001 - DTR, DTR check form 57-31-02-1, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-71.

AIRPLANE NOTE: Applicable to 737-800, -900ER with production installed winglets per Section 9, Table 9-1.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 21000FC.

ACCESS NOTE: Removal of winglet and tension bolts is required.

57-655-00-02	AWL	57-05-02-250-869	1.1	NOTE		800 900ER	ALL

Inspect (Open Hole Eddy Current) all eighteen upper and lower tension bolt holes common to the winglet STA 0 Rib.
See Doc. D626A001 - DTR, DTR check form 57-31-02-1, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-71.

AIRPLANE NOTE: Applicable to 737-800, -900ER with production installed winglets per Section 9, Table 9-1.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 21000FC.

ACCESS NOTE: Removal of winglet and tension bolts is required.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-656-00-01	AWL	57-05-02-250-871	1.1	NOTE		800 900ER	ALL

Inspect (Open Hole Eddy Current) the in-spar lower flange of the winglet STA 0 (root) rib.
See Doc. D626A001 - DTR, DTR check form 57-31-02-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-72.

AIRPLANE NOTE: Applicable to 737-800, -900ER with production installed winglets per Section 9, Table 9-1.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 21000FC.

ACCESS NOTE: Removal of winglet and adjacent access panel is required. Remove 2 fasteners forward of the front spar common to the rib flange.

57-656-00-02	AWL	57-05-02-250-871	1.1	NOTE		800 900ER	ALL

Inspect (Open Hole Eddy Current) the in-spar lower flange of the winglet STA 0 (root) rib.
See Doc. D626A001 - DTR, DTR check form 57-31-02-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-10-72.

AIRPLANE NOTE: Applicable to 737-800, -900ER with production installed winglets per Section 9, Table 9-1.

INTERVAL NOTE: Flight length sensitive (FLS) task. See Section 9, Figure 1 to determine threshold. Boeing recommended repeat inspection interval is 21000FC.

ACCESS NOTE: Removal of winglet and adjacent access panel is required. Remove 2 fasteners forward of the front spar common to the rib flange.

57-658-00-01	AWL	57-05-02-211-841	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (Detailed) all the lugs of both support fittings on the inboard actuator support fittings on the main landing gear beam.
See Doc. D626A001 - DTR, DTR check form 57-51-15, for alternative inspection.

ACCESS NOTE: Access requires deployment of inboard flaps.

57-658-00-02	AWL	57-05-02-211-841	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (Detailed) all the lugs of both support fittings on the inboard actuator support fittings on the main landing gear beam.
See Doc. D626A001 - DTR, DTR check form 57-51-15, for alternative inspection.

ACCESS NOTE: Access requires deployment of inboard flaps.

57-659-00-01	AWL	57-05-02-250-873	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the four main lugs on the inboard support fitting of the main landing gear beam.
See Doc. D626A001 - DTR, DTR check form 57-51-16, for alternative inspection.

57-659-00-02	AWL	57-05-02-250-873	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the four main lugs on the inboard support fitting of the main landing gear beam.
See Doc. D626A001 - DTR, DTR check form 57-51-16, for alternative inspection.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-660-00-01	AWL	57-05-02-211-843	1.1	56000 FC	4000 FC	600 700 700C 700IGW 800 900	ALL
	Inspect (Detailed) the outboard main flap, outboard track at WBL 357.7. See Doc. D626A001 - DTR, DTR check form 57-53-01-1a, for alternative inspection. **AIRPLANE NOTE:** All except 900ER						
57-660-00-02	AWL	57-05-02-211-843	1.1	56000 FC	4000 FC	600 700 700C 700IGW 800 900	ALL
	Inspect (Detailed) the outboard main flap, outboard track at WBL 357.7. See Doc. D626A001 - DTR, DTR check form 57-53-01-1a, for alternative inspection. **AIRPLANE NOTE:** All except 900ER						
57-661-00-01	AWL	57-05-02-211-845	1.1	56000 FC	4000 FC	ALL	ALL
	Inspect (Detailed) the catcher attached to the forward fitting assemblies for flap tracks right and left side at WBL 357.7, 254.0 and 164.0 on the inside bottom surface of the catcher. See Doc. D626A001 - DTR, DTR check form 57-53-02, for alternative inspection. **ACCESS NOTE:** Remove catcher. Access to catcher requires removal of the forward fixed flap support fairing.						
57-661-00-02	AWL	57-05-02-211-845	1.1	56000 FC	4000 FC	ALL	ALL
	Inspect (Detailed) the catcher attached to the forward fitting assemblies for flap tracks right and left side at WBL 357.7, 254.0 and 164.0 on the inside bottom surface of the catcher. See Doc. D626A001 - DTR, DTR check form 57-53-02, for alternative inspection. **ACCESS NOTE:** Remove catcher. Access to catcher requires removal of the forward fixed flap support fairing.						
57-662-00-01	AWL	57-05-02-211-847	1.1	56000 FC	4000 FC	600 700 700C 700IGW 800 900	ALL
	Inspect (Detailed) the outboard main flap, inboard track at WBL 254.0. See Doc. D626A001 - DTR, DTR check form 57-53-03-1a, for alternative inspection. **ACCESS NOTE:** For the aft portion of the track the flap must be deployed so the fairing clears the track. For the forward portion of the track the flap support fairing must be removed.						
57-662-00-02	AWL	57-05-02-211-847	1.1	56000 FC	4000 FC	600 700 700C 700IGW 800 900	ALL
	Inspect (Detailed) the outboard main flap, inboard track at WBL 254.0. See Doc. D626A001 - DTR, DTR check form 57-53-03-1a, for alternative inspection. **ACCESS NOTE:** For the aft portion of the track the flap must be deployed so the fairing clears the track. For the forward portion of the track the flap support fairing must be removed.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-664-00-01	AWL	57-05-02-211-849	1.1	56000 FC	4000 FC	600 700 700C 700IGW 800 900	ALL

Inspect (Detailed) the inboard main flap, outboard track at WBL 164.0.
See Doc. D626A001 - DTR, DTR check form 57-53-06-1a, for alternative inspection.

ACCESS NOTE: For the aft portion of the track the flap must be deployed so the fairing clears the track. For the forward portion of the track the flap support fairing must be removed.

57-664-00-02	AWL	57-05-02-211-849	1.1	56000 FC	4000 FC	600 700 700C 700IGW 800 900	ALL

Inspect (Detailed) the inboard main flap, outboard track at WBL 164.0.
See Doc. D626A001 - DTR, DTR check form 57-53-06-1a, for alternative inspection.

ACCESS NOTE: For the aft portion of the track the flap must be deployed so the fairing clears the track. For the forward portion of the track the flap support fairing must be removed.

57-666-00-01	AWL	57-05-02-130-831	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (Ultrasonic) the inboard flap, inboard track aft link pins at WBL 64.0.
See Doc. D626A001 - DTR, DTR check form 57-53-09, for alternative inspection.

ACCESS NOTE: Inner pin removal required.

57-666-00-02	AWL	57-05-02-130-831	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (Ultrasonic) the inboard flap, inboard track aft link pins at WBL 64.0.
See Doc. D626A001 - DTR, DTR check form 57-53-09, for alternative inspection.

ACCESS NOTE: Inner pin removal required.

57-667-00-01	AWL	57-05-02-211-851	1.1	56000 FC	4000 FC	600 700 700C 700IGW 800 900	ALL

Inspect (Detailed) the inboard main flap, inboard track at WBL 64.0.
See Doc. D626A001 - DTR, DTR check form 57-53-10-1a, for alternative inspection.

ACCESS NOTE: Side of body fairing must be removed to gain access. For HFEC inspections the carriage must be cycled to gain access.

57-667-00-02	AWL	57-05-02-211-851	1.1	56000 FC	4000 FC	600 700 700C 700IGW 800 900	ALL

Inspect (Detailed) the inboard main flap, inboard track at WBL 64.0.
See Doc. D626A001 - DTR, DTR check form 57-53-10-1a, for alternative inspection.

ACCESS NOTE: Side of body fairing must be removed to gain access. For HFEC inspections the carriage must be cycled to gain access.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-668-00-01	AWL	57-05-02-250-883	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the primary lug of the inboard main flap forward fitting on the inboard track.
See Doc. D626A001 - DTR, DTR check form 57-53-11, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-50-20.

ACCESS NOTE: Flap track removal required to perform this inspection.

57-668-00-02	AWL	57-05-02-250-883	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the primary lug of the inboard main flap forward fitting on the inboard track.
See Doc. D626A001 - DTR, DTR check form 57-53-11, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-50-20.

ACCESS NOTE: Flap track removal required to perform this inspection.

57-669-00-01	AWL	57-05-02-211-853	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (Detailed) the front spar upper chord trailing edge on the inboard main flap from Inboard Trailing Edge Flap (ITEF) main flap STA 73 to 167, away from the rib.
See Doc. D626A001 - DTR, DTR check form 57-53-12-1, for alternative inspection.

ACCESS NOTE: Access requires removal of inspar skin panel.

57-669-00-02	AWL	57-05-02-211-853	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (Detailed) the front spar upper chord trailing edge on the inboard main flap from Inboard Trailing Edge Flap (ITEF) main flap STA 73 to 167, away from the rib.
See Doc. D626A001 - DTR, DTR check form 57-53-12-1, for alternative inspection.

ACCESS NOTE: Access requires removal of inspar skin panel.

57-669-10-01	AWL	57-05-02-250-885	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the front spar upper and lower chord on the trailing edge and the inboard main flap from Inboard Trailing Edge Flap (ITEF) main flap STA 73 to 167, at the rib location.
See Doc. D626A001 - DTR, DTR check form 57-53-12-2, for alternative inspection.

57-669-10-02	AWL	57-05-02-250-885	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the front spar upper and lower chord on the trailing edge and the inboard main flap from Inboard Trailing Edge Flap (ITEF) main flap STA 73 to 167, at the rib location.
See Doc. D626A001 - DTR, DTR check form 57-53-12-2, for alternative inspection.

57-669-20-01	AWL	57-05-02-210-820	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (General Visual) the front spar cutout /front spar fitting on the trailing edge, inboard main flap from Inboard Trailing Edge Flap (ITEF) main flap STA 155.00.
See Doc. D626A001 - DTR, DTR check form 57-53-12-3, for alternative inspection.

ACCESS NOTE: Nose skin over the cutout must be removed, and flaps deployed.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-669-20-02	AWL	57-05-02-210-820	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (General Visual) the front spar cutout /front spar fitting on the trailing edge, inboard main flap from Inboard Trailing Edge Flap (ITEF) main flap STA 155.00. See Doc. D626A001 - DTR, DTR check form 57-53-12-3, for alternative inspection. **ACCESS NOTE:** Nose skin over the cutout must be removed, and flaps deployed.						
57-670-00-01	AWL	57-05-02-211-855	1.1	56000 FC	9000 FC	ALL	ALL
	Inspect (Detailed) the upper and lower rear spar chord at the inboard and outboard cutout on the trailing edge inboard main flap at Inboard Trailing Edge Flap (ITEF) and main flap STA 85.225 and 143.3. See Doc. D626A001 - DTR, DTR check form 57-53-13-1, for alternative inspection. **ACCESS NOTE:** Lower inspar skin panel removal is required.						
57-670-00-02	AWL	57-05-02-211-855	1.1	56000 FC	9000 FC	ALL	ALL
	Inspect (Detailed) the upper and lower rear spar chord at the inboard and outboard cutout on the trailing edge inboard main flap at Inboard Trailing Edge Flap (ITEF) and main flap STA 85.225 and 143.3. See Doc. D626A001 - DTR, DTR check form 57-53-13-1, for alternative inspection. **ACCESS NOTE:** Lower inspar skin panel removal is required.						
57-670-10-01	AWL	57-05-02-211-857	1.1	56000 FC	9000 FC	ALL	ALL
	Inspect (Detailed) the horizontal and vertical flanges of the rear spar upper and lower chords at the inboard and outboard pushrod cutout on the trailing edge, inboard main flap at Inboard Trailing Edge Flap (ITEF) main flap STA 91.5 and 149.0. See Doc. D626A001 - DTR, DTR check form 57-53-13-2, for alternative inspection. **ACCESS NOTE:** Skin panel removal is required.						
57-670-10-02	AWL	57-05-02-211-857	1.1	56000 FC	9000 FC	ALL	ALL
	Inspect (Detailed) the horizontal and vertical flanges of the rear spar upper and lower chords at the inboard and outboard pushrod cutout on the trailing edge, inboard main flap at Inboard Trailing Edge Flap (ITEF) main flap STA 91.5 and 149.0. See Doc. D626A001 - DTR, DTR check form 57-53-13-2, for alternative inspection. **ACCESS NOTE:** Skin panel removal is required.						
57-670-11-01	AWL	57-05-02-211-858	1.1	56000 FC	9000 FC	ALL	ALL
	Inspect (Detailed) the horizontal and vertical flanges of the rear spar upper and lower chords at the inboard and outboard pushrod cutout on the trailing edge, inboard main flap at Inboard Trailing Edge Flap (ITEF) main flap STA 91.5 and 149.0. See Doc. D626A001 - DTR, DTR check form 57-53-13-2, for alternative inspection. **ACCESS NOTE:** Flap deployment is required.						
57-670-11-02	AWL	57-05-02-211-858	1.1	56000 FC	9000 FC	ALL	ALL
	Inspect (Detailed) the horizontal and vertical flanges of the rear spar upper and lower chords at the inboard and outboard pushrod cutout on the trailing edge, inboard main flap at Inboard Trailing Edge Flap (ITEF) main flap STA 91.5 and 149.0. See Doc. D626A001 - DTR, DTR check form 57-53-13-2, for alternative inspection. **ACCESS NOTE:** Flap deployment is required.						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-670-20-01	AWL	57-05-02-211-861	1.1	56000 FC	9000 FC	ALL	ALL

Inspect (Detailed) the rear spar upper chord on the trailing edge, inboard main flap from Inboard Trailing Edge Flap (ITEF) and main flap STA 73 to 167, away from the rib.
See Doc. D626A001 - DTR, DTR check form 57-53-13-3, for alternative inspection.

ACCESS NOTE: Removal of skin panels is required.

57-670-20-02	AWL	57-05-02-211-861	1.1	56000 FC	9000 FC	ALL	ALL

Inspect (Detailed) the rear spar upper chord on the trailing edge, inboard main flap from Inboard Trailing Edge Flap (ITEF) and main flap STA 73 to 167, away from the rib.
See Doc. D626A001 - DTR, DTR check form 57-53-13-3, for alternative inspection.

ACCESS NOTE: Removal of skin panels is required.

57-670-30-01	AWL	57-05-02-250-887	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the rear spar upper chord on the trailing edge, inboard main flap from Inboard Trailing Edge Flap (ITEF) and main flap STA 73 to 167 at the rib.
See Doc. D626A001 - DTR, DTR check form 57-53-13-4, for alternative inspection.

57-670-30-02	AWL	57-05-02-250-887	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the rear spar upper chord on the trailing edge, inboard main flap from Inboard Trailing Edge Flap (ITEF) and main flap STA 73 to 167 at the rib.
See Doc. D626A001 - DTR, DTR check form 57-53-13-4, for alternative inspection.

57-671-00-01	AWL	57-05-02-211-863	1.1	56000 FC	9000 FC	ALL	ALL

Inspect (Detailed) the skin at the rear spar chord, inboard main flap from Inboard Trailing Edge Flap (ITEF) and main flap STA 73 to 167.
See Doc. D626A001 - DTR, DTR check form 57-53-14, for alternative inspection.

57-671-00-02	AWL	57-05-02-211-863	1.1	56000 FC	9000 FC	ALL	ALL

Inspect (Detailed) the skin at the rear spar chord, inboard main flap from Inboard Trailing Edge Flap (ITEF) and main flap STA 73 to 167.
See Doc. D626A001 - DTR, DTR check form 57-53-14, for alternative inspection.

57-672-00-01	AWL	57-05-02-250-889	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the inboard main flap, inboard torque tube rib on the lower chord.
See Doc. D626A001 - DTR, DTR check form 57-53-17, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-50-27.

ACCESS NOTE: Removal of flap lower skin is required for inspection.

57-672-00-02	AWL	57-05-02-250-889	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the inboard main flap, inboard torque tube rib on the lower chord.
See Doc. D626A001 - DTR, DTR check form 57-53-17, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-50-27.

ACCESS NOTE: Removal of flap lower skin is required for inspection.

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-673-00-01	AWL	57-05-02-210-822	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (General Visual) the inboard main flap torque tube on all of the exposed torque tube surfaces. See Doc. D626A001 - DTR, DTR check form 57-53-20, for alternative inspection.						
57-673-00-02	AWL	57-05-02-210-822	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (General Visual) the inboard main flap torque tube on all of the exposed torque tube surfaces. See Doc. D626A001 - DTR, DTR check form 57-53-20, for alternative inspection.						
57-674-00-01	AWL	57-05-02-250-891	1.1	56000 FC	36000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) all fitting cutout edge surfaces on the inboard main flap, rear spar aft flap track support assemblies at the Inboard Trailing Edge Flap (ITEF) and main flap STA 85 and 143. See Doc. D626A001 - DTR, DTR check form 57-53-21, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-50-28. **ACCESS NOTE:** Lower skin panel must be removed and the aft flap track pushrods disconnected to allow the aft flap tracks to extend to their stops.						
57-674-00-02	AWL	57-05-02-250-891	1.1	56000 FC	36000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) all fitting cutout edge surfaces on the inboard main flap, rear spar aft flap track support assemblies at the Inboard Trailing Edge Flap (ITEF) and main flap STA 85 and 143. See Doc. D626A001 - DTR, DTR check form 57-53-21, for alternative inspection. The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-50-28. **ACCESS NOTE:** Lower skin panel must be removed and the aft flap track pushrods disconnected to allow the aft flap tracks to extend to their stops.						
57-675-00-01	AWL	57-05-02-250-893	1.1	56000 FC	36000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the forward and aft lug on the carriage assembly. See Doc. D626A001 - DTR, DTR check form 57-53-22, for alternative inspection. **ACCESS NOTE:** The flaps must be deployed to gain inspection access. In addition, the roller assembly must be removed to gain access to the Inbd and Outbd side surface of the roller lug.						
57-675-00-02	AWL	57-05-02-250-893	1.1	56000 FC	36000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the forward and aft lug on the carriage assembly. See Doc. D626A001 - DTR, DTR check form 57-53-22, for alternative inspection. **ACCESS NOTE:** The flaps must be deployed to gain inspection access. In addition, the roller assembly must be removed to gain access to the Inbd and Outbd side surface of the roller lug.						
57-676-00-01	AWL	57-05-02-250-895	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (High Frequency Eddy Current) the forward and aft lugs on the carriage plates. See Doc. D626A001 - DTR, DTR check form 57-53-23-1,alternative inspection. **ACCESS NOTE:** The side of body fairing must be removed to gain access. In addition, the aft roller cartridge assembly must be removed to gain access to the inboard and outboard side surfaces of the aft roller lug.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-676-00-02	AWL	57-05-02-250-895	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the forward and aft lugs on the carriage plates.
See Doc. D626A001 - DTR, DTR check form 57-53-23-1,alternative inspection.

ACCESS NOTE: The side of body fairing must be removed to gain access. In addition, the aft roller cartridge assembly must be removed to gain access to the inboard and outboard side surfaces of the aft roller lug.

57-676-10-01	AWL	57-05-02-211-865	1.1	NOTE		ALL	ALL

Inspect (Detailed) the torque tube attachment hoop holes on the carriage plate.
See Doc. D626A001 - DTR, DTR check form 57-53-23-2, alternative inspection.

INTERVAL NOTE: 56000 FC threshold applicable to all airplanes (L/N 4021 and on) and those airplanes (L/N 1-4020) that have incorporated SB 737-57A-1314. Repeat interval is 18000 FC.

32000 FC threshold applicable to airplanes (L/N 1-4020) that have not incorporated SB 737-57A-1314. Repeat interval is 18000 FC.

57-676-10-02	AWL	57-05-02-211-865	1.1	NOTE		ALL	ALL

Inspect (Detailed) the torque tube attachment hoop holes on the carriage plate.
See Doc. D626A001 - DTR, DTR check form 57-53-23-2, alternative inspection.

INTERVAL NOTE: 56000 FC threshold applicable to all airplanes (L/N 4021 and on) and those airplanes (L/N 1-4020) that have incorporated SB 737-57A-1314. Repeat interval is 18000 FC.

32000 FC threshold applicable to airplanes (L/N 1-4020) that have not incorporated SB 737-57A-1314. Repeat interval is 18000 FC.

57-676-20-01	AWL	57-05-02-211-867	1.1	56000 FC	9000 FC	ALL	ALL

Inspect (Detailed) the aft upper chord, rub pad hole on the inboard carriage.
See Doc. D626A001 - DTR, DTR check form 57-53-23-3, for alternative inspection.

ACCESS NOTE: Removal of rub pad attachment fastener is required.

57-676-20-02	AWL	57-05-02-211-867	1.1	56000 FC	9000 FC	ALL	ALL

Inspect (Detailed) the aft upper chord, rub pad hole on the inboard carriage.
See Doc. D626A001 - DTR, DTR check form 57-53-23-3, for alternative inspection.

ACCESS NOTE: Removal of rub pad attachment fastener is required.

57-677-00-01	AWL	57-05-02-250-897	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the four outboard carriage roller bosses on the outboard main flap at WBL 357.7.
See Doc. D626A001 - DTR, DTR check form 57-53-24-1, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-50-23.

ACCESS NOTE: Flaps must be deployed and roller pin assemblies removed.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-677-00-02	AWL	57-05-02-250-897	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the four outboard carriage roller bosses on the outboard main flap at WBL 357.7.
See Doc. D626A001 - DTR, DTR check form 57-53-24-1, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-50-23.

ACCESS NOTE: Flaps must be deployed and roller pin assemblies removed.

57-677-10-01	AWL	57-05-02-250-899	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the holes in the carriage which attach the aft bridge fitting at the outboard carriage aft bridge support on the outboard main flap at WBL 357.7.
See Doc. D626A001 - DTR, DTR check form 57-53-24-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-50-23.

ACCESS NOTE: The flaps must be deployed and the aft bridge fitting and bolts must be removed for inspection.

57-677-10-02	AWL	57-05-02-250-899	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the holes in the carriage which attach the aft bridge fitting at the outboard carriage aft bridge support on the outboard main flap at WBL 357.7.
See Doc. D626A001 - DTR, DTR check form 57-53-24-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-50-23.

ACCESS NOTE: The flaps must be deployed and the aft bridge fitting and bolts must be removed for inspection.

57-678-00-01	AWL	57-05-02-250-901	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the four inboard carriage roller bosses on the outboard main flap at WBL 254.0.
See Doc. D626A001 - DTR, DTR check form 57-53-25-1, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-50-24.

ACCESS NOTE: The flaps must be deployed and roller pin assemblies removed.

57-678-00-02	AWL	57-05-02-250-901	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the four inboard carriage roller bosses on the outboard main flap at WBL 254.0.
See Doc. D626A001 - DTR, DTR check form 57-53-25-1, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-50-24.

ACCESS NOTE: The flaps must be deployed and roller pin assemblies removed.

57-678-10-01	AWL	57-05-02-250-903	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the holes in the carriage which attach the aft bridge fitting on the outboard main flap at WBL 254.0.
See Doc. D626A001 - DTR, DTR check form 57-53-25-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-50-24.

ACCESS NOTE: The flaps must be deployed and the aft bridge fitting and bolts must be removed for the inspection.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-678-10-02	AWL	57-05-02-250-903	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the holes in the carriage which attach the aft bridge fitting on the outboard main flap at WBL 254.0.
See Doc. D626A001 - DTR, DTR check form 57-53-25-2, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Section 57-50-24.

ACCESS NOTE: The flaps must be deployed and the aft bridge fitting and bolts must be removed for the inspection.

57-679-00-01	AWL	57-05-02-250-905	1.1	56000 FC	9000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the spar upper chord at the ribs from WBL 280 to WBL 327 and the spar lower chord at the ribs from WBL 254 to WBL 358.
See Doc. D626A001 - DTR, DTR check form 57-53-26-1, for alternative inspection.

ACCESS NOTE: Deployment of flaps provides access at the carriage support ribs.

57-679-00-02	AWL	57-05-02-250-905	1.1	56000 FC	9000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the spar upper chord at the ribs from WBL 280 to WBL 327 and the spar lower chord at the ribs from WBL 254 to WBL 358.
See Doc. D626A001 - DTR, DTR check form 57-53-26-1, for alternative inspection.

ACCESS NOTE: Deployment of flaps provides access at the carriage support ribs.

57-680-00-01	AWL	57-05-02-250-907	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the spar upper chord between the ribs from WBL 280 to WBL 327 and the spar lower chord between the ribs from WBL 254 to WBL 358.
See Doc. D626A001 - DTR, DTR check form 57-53-26-2, for alternative inspection.

ACCESS NOTE: Removal of the skin is required.

57-680-00-02	AWL	57-05-02-250-907	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the spar upper chord between the ribs from WBL 280 to WBL 327 and the spar lower chord between the ribs from WBL 254 to WBL 358.
See Doc. D626A001 - DTR, DTR check form 57-53-26-2, for alternative inspection.

ACCESS NOTE: Removal of the skin is required.

57-681-00-01	AWL	57-05-02-250-909	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the aft flap track cutouts 2 & 3 at the rear spar outboard main flap.
See Doc. D626A001 - DTR, DTR check form 57-53-27-1, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-50-21.

ACCESS NOTE: Internal access is required. Removal of the flap upper skin is required to gain access. The aft flap track pushrods need to be disconnected to allow the aft flap tracks to extend to their stops.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-681-00-02	AWL	57-05-02-250-909	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the aft flap track cutouts 2 & 3 at the rear spar outboard main flap.
See Doc. D626A001 - DTR, DTR check form 57-53-27-1, for alternative inspection.
The NDI method(s) necessary to accomplish the intent of this inspection is contained in the 737 Nondestructive Test Manual (D6-37239). The inspection procedures are contained in Part 6, Subject 57-50-21.

ACCESS NOTE: Internal access is required. Removal of the flap upper skin is required to gain access. The aft flap track pushrods need to be disconnected to allow the aft flap tracks to extend to their stops.

57-682-00-01	AWL	57-05-02-250-911	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the rear spar upper chord at the ribs from WBL 280 to WBL 327 and the rear spar lower chord at the ribs from WBL 254 to WBL 358.
See Doc. D626A001 - DTR, DTR check form 57-53-27-2, for alternative inspection.

ACCESS NOTE: Deployment of flaps provides access at the carriage support ribs.

57-682-00-02	AWL	57-05-02-250-911	1.1	56000 FC	18000 FC	ALL	ALL

Inspect (Low Frequency Eddy Current) the rear spar upper chord at the ribs from WBL 280 to WBL 327 and the rear spar lower chord at the ribs from WBL 254 to WBL 358.
See Doc. D626A001 - DTR, DTR check form 57-53-27-2, for alternative inspection.

ACCESS NOTE: Deployment of flaps provides access at the carriage support ribs.

57-683-00-01	AWL	57-05-02-250-913	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the rear spar upper chord between the ribs from WBL 280 to WBL 327 and the rear spar lower chord between the ribs from WBL 254 to WBL 358.
See Doc. D626A001 - DTR, DTR check form 57-53-27-3, for alternative inspection.

ACCESS NOTE: Removal of flap upper skin panel is required for access.

57-683-00-02	AWL	57-05-02-250-913	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the rear spar upper chord between the ribs from WBL 280 to WBL 327 and the rear spar lower chord between the ribs from WBL 254 to WBL 358.
See Doc. D626A001 - DTR, DTR check form 57-53-27-3, for alternative inspection.

ACCESS NOTE: Removal of flap upper skin panel is required for access.

57-684-00-01	AWL	57-05-02-250-915	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the external surface of the outboard main flap lower skin at the skin-to-chord fastener locations at both the front and rear spar.
See Doc. D626A001 - DTR, DTR check form 57-53-29, for alternative inspection.

ACCESS NOTE: Deployment of flaps provides access at the carriage support ribs.

57-684-00-02	AWL	57-05-02-250-915	1.1	56000 FC	36000 FC	ALL	ALL

Inspect (High Frequency Eddy Current) the external surface of the outboard main flap lower skin at the skin-to-chord fastener locations at both the front and rear spar.
See Doc. D626A001 - DTR, DTR check form 57-53-29, for alternative inspection.

ACCESS NOTE: Deployment of flaps provides access at the carriage support ribs.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-685-00-01	AWL	57-05-02-240-801	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (Magnetic Particle) the front spar spigot fitting at the surface of the fitting shaft at WBL 254.0 and WBL 358.0. See Doc. D626A001 - DTR, DTR check form PSE 57-53-31, for alternative inspection. **ACCESS NOTE:** Access requires removal of the flap from the carriages.						
57-685-00-02	AWL	57-05-02-240-801	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (Magnetic Particle) the front spar spigot fitting at the surface of the fitting shaft at WBL 254.0 and WBL 358.0. See Doc. D626A001 - DTR, DTR check form PSE 57-53-31, for alternative inspection. **ACCESS NOTE:** Access requires removal of the flap from the carriages.						
57-686-00-01	AWL	57-05-02-211-877	1.1	56000 FC	9000 FC	ALL	ALL
	Inspect (Detailed) the carriage aft link assembly (aft spigot fitting, aft link, aft bridge fitting clevis, aft pin) at WBL 254.0 and 358.0. See Doc. D626A001-DTR, DTR check form 57-53-32, for alternative inspections. **ACCESS NOTE:** Flap deployment is required to gain access.						
57-686-00-02	AWL	57-05-02-211-877	1.1	56000 FC	9000 FC	ALL	ALL
	Inspect (Detailed) the carriage aft link assembly (aft spigot fitting, aft link, aft bridge fitting clevis, aft pin) at WBL 254.0 and 358.0. See Doc. D626A001-DTR, DTR check form 57-53-32, for alternative inspections. **ACCESS NOTE:** Flap deployment is required to gain access.						
57-687-00-01	AWL	57-05-02-211-871	1.1	56000 FC	9000 FC	ALL	ALL
	Inspect (Detailed) the lower flanges of ribs #2 and #3 on each aft flap track support ribs from WBL 254.0 and 358.0. See Doc. D626A001 - DTR, DTR check form 57-53-35, for alternative inspection. **ACCESS NOTE:** Flaps deployed and the lower cove panels removed for access.						
57-687-00-02	AWL	57-05-02-211-871	1.1	56000 FC	9000 FC	ALL	ALL
	Inspect (Detailed) the lower flanges of ribs #2 and #3 on each aft flap track support ribs from WBL 254.0 and 358.0. See Doc. D626A001 - DTR, DTR check form 57-53-35, for alternative inspection. **ACCESS NOTE:** Flaps deployed and the lower cove panels removed for access.						
57-688-00-01	AWL	57-05-02-211-873	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (Detailed) the lugs on both hinge/actuator fittings on the inboard spoiler. See Doc. D626A001 - DTR, DTR check form 57-70-01, for alternative inspection. **ACCESS NOTE:** Access requires deployment of inboard flap.						
57-688-00-02	AWL	57-05-02-211-873	1.1	56000 FC	18000 FC	ALL	ALL
	Inspect (Detailed) the lugs on both hinge/actuator fittings on the inboard spoiler. See Doc. D626A001 - DTR, DTR check form 57-70-01, for alternative inspection. **ACCESS NOTE:** Access requires deployment of inboard flap.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-800-00-01	MRB	05-41-01-210-816	1.1 1.2	36000 FC 10 YR	36000 FC 10 YR	ALL	ALL
	Perform an internal zonal inspection (GV) of the center section wing box - section 44, sta 540 to sta 663.75. **INTERVAL NOTE:** Whichever comes first.						
57-802-01-01	MRB	05-41-05-210-801	1.1	120 DY	120 DY	ALL	ALL
	Perform an external zonal inspection (GV) of the left wing. Inspection is accomplished from the ground, without the use of stands or ladders. No additional access panels required. **ACCESS NOTE:** Control surfaces extended.						
57-804-01-01	MRB	05-41-05-210-804	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the leading edge to front spar - inboard of nacelle strut - left wing. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Excluding surface under wing to body fairing.						
57-806-01-01	MRB	05-41-05-210-805	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the leading edge to front spar - inboard of nacelle strut - left wing. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 18000 FC/6 YR is satisfied by this zonal inspection. **ACCESS NOTE:** Flaps deployed.						
57-808-01-01	MRB	05-41-05-210-806	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the Krueger flaps no. 1 and 2 - left wing. **INTERVAL NOTE:** Whichever comes first.						
57-810-01-01	MRB	05-41-05-210-807	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the Krueger flaps no. 1 and 2 - left wing. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Flaps deployed.						
57-812-01-01	MRB	05-41-05-210-808	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the leading edge to front spar - outboard of nacelle strut - left wing. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Slats deployed.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-814-01-01	MRB	05-41-05-210-809	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the leading edge to front spar - outboard of nacelle strut - left wing. (EZAP)						
	INTERVAL NOTE: Whichever comes first. The EZAP inspection requirement with interval 36000 FC/12 YR is satisfied by this zonal inspection.						
	ACCESS NOTE: Slats deployed.						
57-816-01-01	MRB	05-41-05-210-810	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the slats no. 1, 2, 3, 4 - left wing.						
	INTERVAL NOTE: Whichever comes first.						
	ACCESS NOTE: Slats in full extend position.						
57-822-01-01	MRB	05-41-05-210-813	1.1 1.2	36000 FC 10 YR	36000 FC 10 YR	ALL	ALL
	Perform an internal zonal inspection (GV) of the center fuel tank - left wing.						
	INTERVAL NOTE: Whichever comes first.						
57-824-01-01	MRB	05-41-05-210-814	1.1 1.2	5500 FC 30 MO	5500 FC 30 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the center fuel tank - left wing.						
	INTERVAL NOTE: Whichever comes first.						
	ACCESS NOTE: Excluding surface under wing to body fairing.						
57-826-01-01	MRB	05-41-05-210-815	1.1 1.2	36000 FC 10 YR	36000 FC 10 YR	ALL	ALL
	Perform an internal zonal inspection (GV) of the main fuel tank - left wing.						
	INTERVAL NOTE: Whichever comes first.						
	ACCESS NOTE: Caution - Do not remove 532AZ and 532BZ at the same time.						
57-828-01-01	MRB	05-41-05-210-816	1.1 1.2	5500 FC 30 MO	5500 FC 30 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the main fuel tank - left wing.						
	INTERVAL NOTE: Whichever comes first.						
57-830-01-01	MRB	05-41-05-210-817	1.1 1.2	36000 FC 10 YR	36000 FC 10 YR	ALL	ALL
	Perform an internal zonal inspection (GV) of the surge tank - left wing.						
	INTERVAL NOTE: Whichever comes first.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-832-01-01	MRB	05-41-05-210-818	1.1 1.2	5500 FC 30 MO	5500 FC 30 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the surge tank - left wing. **INTERVAL NOTE:** Whichever comes first.						
57-834-01-01	MRB	05-41-05-210-819	1.1 1.2	5500 FC 30 MO	5500 FC 30 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the dry bay - left wing. **INTERVAL NOTE:** Whichever comes first.						
57-836-01-01	MRB	05-41-05-210-821	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the fairing flap support no. 3 - left wing. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Flaps extended.						
57-838-01-01	MRB	05-41-05-210-822	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the fairing flap support no. 3 - left wing. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 18000 FC/6 YR is satisfied by this zonal inspection. **ACCESS NOTE:** Flaps extended, inside of fairing.						
57-840-01-01	MRB	05-41-05-210-823	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the fairing flap support no. 2 - left wing. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Flaps extended.						
57-842-01-01	MRB	05-41-05-210-824	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the fairing flap support no. 2 - left wing. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 18000 FC/6 YR is satisfied by this zonal inspection. **ACCESS NOTE:** Flaps extended, inside of fairing.						
57-844-01-01	MRB	05-41-05-210-825	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the fairing flap support no. 1 - left wing. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Flaps extended.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-846-01-01	MRB	05-41-05-210-826	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the fairing flap support no. 1 - left wing. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 18000 FC/6 YR is satisfied by this zonal inspection. **ACCESS NOTE:** Flaps extended, inside of fairing.						
57-848-01-01	MRB	05-41-05-210-827	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the rear spar to landing gear support beam - left wing. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Excluding surface under wing to body fairing.						
57-850-01-01	MRB	05-41-05-210-828	1.1 1.2	21600 FC 6 YR	21600 FC 6 YR	ALL	ALL
	Perform an internal zonal inspection (GV) of the rear spar to landing gear support beam - left wing. **INTERVAL NOTE:** Whichever comes first.						
57-852-01-01	MRB	05-41-05-210-829	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the inboard spoiler no. 6 - left wing. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Spoiler raised.						
57-854-01-01	MRB	05-41-05-210-830	1.1 1.2	5500 FC 24 MO	5500 FC 24 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the inboard flaps - left wing. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Flaps extended.						
57-856-01-01	MRB	05-41-05-210-831	1.1 1.2	21600 FC 6 YR	21600 FC 6 YR	ALL	ALL
	Perform an internal zonal inspection (GV) of the inboard main flap - left wing. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Flaps extended.						
57-858-01-01	MRB	05-41-05-210-832	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the rear spar to trailing edge - outboard of inboard flap - inboard of fixed trailing edge - left wing. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 18000 FC/6 YR is satisfied by this zonal inspection. **ACCESS NOTE:** Flaps down, spoilers raised.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-860-01-01	MRB	05-41-05-210-833	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the spoilers 1, 2, 3, 4, 5 - left wing. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Spoilers raised.						
57-862-01-01	MRB	05-41-05-210-834	1.1 1.2	5500 FC 30 MO	5500 FC 30 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the outboard flaps - left wing. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Flaps extended.						
57-864-01-01	MRB	05-41-05-210-835	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the fixed trailing edge - left wing. **INTERVAL NOTE:** Whichever comes first.						
57-866-01-01	MRB	05-41-05-210-836	1.1	10 YR	10 YR	ALL	ALL
	Perform an internal zonal inspection (GV) of the fixed trailing edge - left wing.						
57-868-01-01	MRB	05-41-05-210-837	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the fixed trailing edge - left wing. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 36000 FC/12 YR is satisfied by this zonal inspection. **ACCESS NOTE:** Aileron control area only.						
57-870-01-01	MRB	05-41-05-210-838	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the aileron - left wing. **INTERVAL NOTE:** Whichever comes first.						
57-872-02-01	MRB	05-41-06-210-801	1.1	120 DY	120 DY	ALL	ALL
	Perform an external zonal inspection (GV) of the right wing. Inspection is accomplished from the ground, without the use of stands or ladders. No additional access panels required. **ACCESS NOTE:** Control surfaces extended.						
57-874-02-01	MRB	05-41-06-210-804	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the leading edge to front spar - inboard of the nacelle strut - right wing. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Excluding surface under wing to body fairing.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-876-02-01	MRB	05-41-06-210-805	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL

Perform an internal zonal inspection (GV) of the leading edge to front spar - inboard of the nacelle strut - right wing. (EZAP)

INTERVAL NOTE: Whichever comes first. The EZAP inspection requirement with interval 18000 FC/6 YR is satisfied by this zonal inspection.

ACCESS NOTE: Flaps deployed.

57-878-02-01	MRB	05-41-06-210-806	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL

Perform an external zonal inspection (GV) of the Krueger flaps no. 3 and 4 - right wing.

INTERVAL NOTE: Whichever comes first.

57-880-02-01	MRB	05-41-06-210-807	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL

Perform an internal zonal inspection (GV) of the Krueger flaps no. 3 and 4 - right wing.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Flaps deployed.

57-882-02-01	MRB	05-41-06-210-808	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL

Perform an external zonal inspection (GV) of the leading edge to front spar - outboard of the nacelle strut - right wing.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Slats deployed.

57-884-02-01	MRB	05-41-06-210-809	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL

Perform an internal zonal inspection (GV) of the leading edge to front spar - outboard of the nacelle strut - right wing. (EZAP)

INTERVAL NOTE: Whichever comes first. The EZAP inspection requirement with interval 36000 FC/12 YR is satisfied by this zonal inspection.

ACCESS NOTE: Slats deployed.

57-886-02-01	MRB	05-41-06-210-810	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL

Perform an external zonal inspection (GV) of the slats no. 5, 6, 7, 8 - right wing.

INTERVAL NOTE: Whichever comes first.

ACCESS NOTE: Slats in full extend position.

57-892-02-01	MRB	05-41-06-210-813	1.1 1.2	36000 FC 10 YR	36000 FC 10 YR	ALL	ALL

Perform an internal zonal inspection (GV) of the center fuel tank - right wing.

INTERVAL NOTE: Whichever comes first.

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-894-02-01	MRB	05-41-06-210-814	1.1 1.2	5500 FC 30 MO	5500 FC 30 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the center fuel tank - right wing. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Excluding surface under wing to body fairing.						
57-896-02-01	MRB	05-41-06-210-815	1.1 1.2	36000 FC 10 YR	36000 FC 10 YR	ALL	ALL
	Perform an internal zonal inspection (GV) of the main fuel tank - right wing. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Caution - Do not remove 632AZ and 632BZ at the same time.						
57-898-02-01	MRB	05-41-06-210-816	1.1 1.2	5500 FC 30 MO	5500 FC 30 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the main fuel tank - right wing. **INTERVAL NOTE:** Whichever comes first.						
57-900-02-01	MRB	05-41-06-210-817	1.1 1.2	36000 FC 10 YR	36000 FC 10 YR	ALL	ALL
	Perform an internal zonal inspection (GV) of the surge tank - right wing. **INTERVAL NOTE:** Whichever comes first.						
57-902-02-01	MRB	05-41-06-210-818	1.1 1.2	5500 FC 30 MO	5500 FC 30 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the surge tank - right wing. **INTERVAL NOTE:** Whichever comes first.						
57-904-02-01	MRB	05-41-06-210-819	1.1 1.2	5500 FC 30 MO	5500 FC 30 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the dry bay - right wing. **INTERVAL NOTE:** Whichever comes first.						
57-906-02-01	MRB	05-41-06-210-821	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the fairing flap support no. 6 - right wing. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Flaps extended.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-908-02-01	MRB	05-41-06-210-822	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the fairing flap support no. 6 - right wing. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 18000 FC/6 YR is satisfied by this zonal inspection. **ACCESS NOTE:** Flaps extended, inside of fairing.						
57-910-02-01	MRB	05-41-06-210-823	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the fairing flap support no. 7 - right wing. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Flaps extended.						
57-912-02-01	MRB	05-41-06-210-824	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the fairing flap support no. 7 - right wing. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 18000 FC/6 YR is satisfied by this zonal inspection. **ACCESS NOTE:** Flaps extended, inside of fairing.						
57-914-02-01	MRB	05-41-06-210-825	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the fairing flap support no. 8 - right wing. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Flaps extended.						
57-916-02-01	MRB	05-41-06-210-826	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the fairing flap support no. 8 - right wing. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 18000 FC/6 YR is satisfied by this zonal inspection. **ACCESS NOTE:** Flaps extended, inside of fairing.						
57-918-02-01	MRB	05-41-06-210-827	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the rear spar to landing gear support beam - right wing. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Excluding surface under wing to body fairing.						
57-920-02-01	MRB	05-41-06-210-828	1.1 1.2	21600 FC 6 YR	21600 FC 6 YR	ALL	ALL
	Perform an internal zonal inspection (GV) of the rear spar to landing gear support beam - right wing. **INTERVAL NOTE:** Whichever comes first.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-922-02-01	MRB	05-41-06-210-829	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the inboard spoiler no. 7 - right wing. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Spoiler raised.						
57-924-02-01	MRB	05-41-06-210-830	1.1 1.2	5500 FC 24 MO	5500 FC 24 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the inboard flaps - right wing. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Flaps extended.						
57-926-02-01	MRB	05-41-06-210-831	1.1 1.2	21600 FC 6 YR	21600 FC 6 YR	ALL	ALL
	Perform an internal zonal inspection (GV) of the inboard main flap - right wing. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Flaps extended.						
57-928-02-01	MRB	05-41-06-210-832	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the rear spar to trailing edge - outboard of inboard flap - inboard of fixed trailing edge - right wing. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 18000 FC/6 YR is satisfied by this zonal inspection. **ACCESS NOTE:** Flaps down, spoilers raised.						
57-930-02-01	MRB	05-41-06-210-833	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of spoilers no. 8, 9, 10, 11, 12 - right wing. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Spoilers Raised						
57-932-02-01	MRB	05-41-06-210-834	1.1 1.2	5500 FC 30 MO	5500 FC 30 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the outboard flaps - right wing. **INTERVAL NOTE:** Whichever comes first. **ACCESS NOTE:** Flaps extended.						
57-934-02-01	MRB	05-41-06-210-835	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the fixed trailing edge - right wing. **INTERVAL NOTE:** Whichever comes first.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
57-936-02-01	MRB	05-41-06-210-836	1.1	10 YR	10 YR	ALL	ALL
	Perform an internal zonal inspection (GV) of the fixed trailing edge - right wing.						
57-938-02-01	MRB	05-41-06-210-837	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an internal zonal inspection (GV) of the fixed trailing edge - right wing. (EZAP) **INTERVAL NOTE:** Whichever comes first. The EZAP inspection requirement with interval 36000 FC/12 YR is satisfied by this zonal inspection. **ACCESS NOTE:** Aileron control area only.						
57-940-02-01	MRB	05-41-06-210-838	1.1 1.2	6600 FC 36 MO	6600 FC 36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the aileron - right wing. **INTERVAL NOTE:** Whichever comes first.						
57-950-01-01	MRB	05-41-05-210-802	1.1	36 MO	36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the winglet - left wing. Note: This task satisfies the requirement of the Airplane Partners Boeing (APB) task 57-942-01-0. **AIRPLANE NOTE:** Applicable only on airplanes with winglet installed.						
57-952-01-01	MRB	05-41-05-210-803	1.1	3 YR	3 YR	ALL	ALL
	Perform an internal zonal inspection (GV) of the winglet - left wing. Note: This task satisfies the requirement of the Airplane Partners Boeing (APB) task 57-944-01-0. **AIRPLANE NOTE:** Applicable only to airplanes with winglet installed.						
57-960-02-01	MRB	05-41-06-210-802	1.1	36 MO	36 MO	ALL	ALL
	Perform an external zonal inspection (GV) of the winglet - right wing. Note: This task satisfies the requirement of the Airplane Partners Boeing (APB) task 57-946-01-0. **AIRPLANE NOTE:** Applicable only to airplanes with winglet installed.						
57-962-02-01	MRB	05-41-06-210-803	1.1	3 YR	3 YR	ALL	ALL
	Perform an internal zonal inspection (GV) of the winglet - right wing. Note: This task satisfies the requirement of the Airplane Partners Boeing (APB) task 57-948-01-0. **AIRPLANE NOTE:** Applicable only to airplanes with winglet installed.						
70-800-01-01	MRB	05-41-04-210-801	1.1	120 DY	120 DY	ALL	ALL
	Perform an external zonal inspection (GV) of engine no. 1 and engine no. 1 strut fairings. Inspection is accomplished from the ground, without the use of stands or ladders. No additional access panels required.						
70-802-01-01	MRB	05-41-04-210-802	1.1	2000 FC	2000 FC	ALL	ALL
	Perform an external zonal inspection (GV) of the engine no. 1. (EZAP) **INTERVAL NOTE:** The EZAP inspection requirement with interval 5500 FC/30 MO is satisfied by this zonal inspection.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
70-804-01-01	MRB	05-41-04-210-803	1.1	6600 FC	6600 FC	ALL	ALL
	Perform an external zonal inspection (GV) of the fan cowl - engine no. 1.						
	ACCESS NOTE: Inner and outer exterior surface.						
70-806-01-01	MRB	05-41-04-210-804	1.1	4000 FC	4000 FC	ALL	ALL
	Perform an external zonal inspection (GV) of the thrust reverser - engine no. 1.						
	ACCESS NOTE: Thrust reverser in stowed and deployed position.						
70-808-01-01	MRB	05-41-04-210-805	1.1	2000 FC	2000 FC	ALL	ALL
	Perform an external zonal inspection (GV) of the thrust reverser - engine no. 1. (EZAP)						
	INTERVAL NOTE: The EZAP inspection requirement with interval 5500 FC/30 MO is satisfied by this zonal inspection.						
	ACCESS NOTE: Inner and outer exterior surfaces.						
70-810-02-01	MRB	05-41-04-210-806	1.1	120 DY	120 DY	ALL	ALL
	Perform an external zonal inspection (GV) of engine no. 2 and engine no. 2 strut fairings. Inspection is accomplished from the ground, without the use of stands or ladders. No additional access panels required.						
70-812-02-01	MRB	05-41-04-210-807	1.1	2000 FC	2000 FC	ALL	ALL
	Perform an external zonal inspection (GV) of the engine no. 2. (EZAP)						
	INTERVAL NOTE: The EZAP inspection requirement with interval 5500 FC/30 MO is satisfied by this zonal inspection.						
70-814-02-01	MRB	05-41-04-210-808	1.1	6600 FC	6600 FC	ALL	ALL
	Perform an external zonal inspection (GV) of the fan cowl - engine no. 2.						
	ACCESS NOTE: Inner and outer exterior surface.						
70-816-02-01	MRB	05-41-04-210-809	1.1	4000 FC	4000 FC	ALL	ALL
	Perform an external zonal inspection (GV) of the thrust reverser - engine no. 2						
	ACCESS NOTE: Thrust reverser in stowed and deployed position.						
70-818-02-01	MRB	05-41-04-210-810	1.1	2000 FC	2000 FC	ALL	ALL
	Perform an external zonal inspection (GV) of the thrust reverser - engine no. 2. (EZAP)						
	INTERVAL NOTE: The EZAP inspection requirement with interval 5500 FC/30 MO is satisfied by this zonal inspection.						
	ACCESS NOTE: Inner and outer exterior surfaces.						
71-010-01-01	MRB	F71-11-01-200-801-F00	1.1	2500 FH	2500 FH	ALL	ALL
	Detailed inspection of the left inlet cowl's inner surface.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
71-010-02-01	MRB	F71-11-01-200-801-F00	1.1	2500 FH	2500 FH	ALL	ALL
	Detailed inspection of the right inlet cowl's inner surface.						
71-040-01-01	MRB	54-55-01-200-801 F71-71-00-700-801-F00	1.1	6 YR	6 YR	ALL	ALL
	Operationally check left engine all drain lines.						
71-040-02-01	MRB	54-55-01-200-801 F71-71-00-700-801-F00	1.1	6 YR	6 YR	ALL	ALL
	Operationally check right engine all drain lines.						
72-020-01-01	MRB	F72-21-00-220-801-F00	1.1	2500 FH	2500 FH	ALL	ALL
	Detailed inspection of left engine inlet and fan blades.						
72-020-02-01	MRB	F72-21-00-220-801-F00	1.1	2500 FH	2500 FH	ALL	ALL
	Detailed inspection of right engine inlet and fan blades.						
72-025-01-01	MRB	F72-21-00-640-801-F00	1.1 1.2	5000 FH 3000 FC	5000 FH 3000 FC	ALL	ALL
	Lubricate Left engine fan blades dovetail. **INTERVAL NOTE:** Whichever comes first.						
72-025-02-01	MRB	F72-21-00-640-801-F00	1.1 1.2	5000 FH 3000 FC	5000 FH 3000 FC	ALL	ALL
	Lubricate Right engine fan blades dovetail. **INTERVAL NOTE:** Whichever comes first.						
72-030-01-01	MRB	F72-99-99-000-801-F00	1.1	LIF LIM		ALL	ALL
	Discard left engine fan disk at life limit.						
72-030-02-01	MRB	F72-99-99-000-801-F00	1.1	LIF LIM		ALL	ALL
	Discard right engine fan disk at life limit.						
72-040-01-01	MRB	F72-99-99-000-802-F00	1.1	LIF LIM		ALL	ALL
	Discard left engine booster spool at life limit. Note: Refer to the engine shop manual, Chapter 5 for life limits.						
72-040-02-01	MRB	F72-99-99-000-802-F00	1.1	LIF LIM		ALL	ALL
	Discard right engine booster spool at life limit. Note: Refer to the engine shop manual, Chapter 5 for life limits.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
72-050-01-01	MRB	F72-99-99-000-803-F00	1.1	LIF LIM		ALL	ALL
	Discard left engine fan shaft at life limit. Note: Refer to the engine shop manual, Chapter 5 for life limits.						
72-050-02-01	MRB	F72-99-99-000-803-F00	1.1	LIF LIM		ALL	ALL
	Discard right engine fan shaft at life limit. Note: Refer to the engine shop manual, Chapter 5 for life limits.						
72-070-01-01	MRB	F72-20-00-210-801-F00 F72-60-00-200-801-F00 F72-63-00-200-801-F00	1.1	15000 FH	15000 FH	ALL	ALL
	Visual check of the left engine accessory gearbox/transfer gearbox mount flanges and fan case and fan frame attachment mounts.						
72-070-02-01	MRB	F72-20-00-210-801-F00 F72-60-00-200-801-F00 F72-63-00-200-801-F00	1.1	15000 FH	15000 FH	ALL	ALL
	Visual check of the right engine accessory gearbox/transfer gearbox mount flanges and fan case and fan frame attachment mounts.						
72-100-01-01	MRB	F72-23-04-200-802-F00	1.1	10000 FC	10000 FC	ALL	ALL
	Visual check of the left engine attachment bolts for the thrust mount fittings.						
72-100-02-01	MRB	F72-23-04-200-802-F00	1.1	10000 FC	10000 FC	ALL	ALL
	Visual check of the right engine attachment bolts for the thrust mount fittings.						
72-110-01-01	MRB	F72-23-04-200-801-F00	1.1	15000 FH	15000 FH	ALL	ALL
	Visual check of the left engine thrust mount fittings.						
72-110-02-01	MRB	F72-23-04-200-801-F00	1.1	15000 FH	15000 FH	ALL	ALL
	Visual check of the right engine thrust mount fittings.						
72-130-01-01	MRB	F72-99-99-000-804-F00	1.1	LIF LIM		ALL	ALL
	Discard left engine stages 1 and 2 spool at manufacturer's life limit.						
72-130-02-01	MRB	F72-99-99-000-804-F00	1.1	LIF LIM		ALL	ALL
	Discard right engine stages 1 and 2 spool at manufacturer's life limit.						
72-140-01-01	MRB	F72-99-99-000-805-F00	1.1	LIF LIM		ALL	ALL
	Discard left engine stage 3 disk at manufacturer's life limit.						
72-140-02-01	MRB	F72-99-99-000-805-F00	1.1	LIF LIM		ALL	ALL
	Discard right engine stage 3 disk at manufacturer's life limit.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
72-150-01-01	MRB	F72-99-99-000-806-F00	1.1	LIF LIM		ALL	ALL
	Discard left engine stages 4 - 9 spool at manufacturer's life limit. Note: Refer to the engine shop manual, chapter 5, for life limits.						
72-150-02-01	MRB	F72-99-99-000-806-F00	1.1	LIF LIM		ALL	ALL
	Discard right engine stages 4 - 9 spool at manufacturer's life limit. Note: Refer to the engine shop manual, chapter 5, for life limits.						
72-160-01-01	MRB	F72-99-99-000-807-F00	1.1	LIF LIM		ALL	ALL
	Discard left engine front shaft at manufacturer's life limit. Note: Refer to the engine shop manual, chapter 5, for life limits.						
72-160-02-01	MRB	F72-99-99-000-807-F00	1.1	LIF LIM		ALL	ALL
	Discard right engine front shaft at manufacturer's life limit. Note: Refer to the engine shop manual, chapter 5, for life limits.						
72-170-01-01	MRB	F72-99-99-000-808-F00	1.1	LIF LIM		ALL	ALL
	Discard left engine rear rotating (CDP) seal at manufacturer's life limit. Note: Refer to the engine shop manual, chapter 5, for life limits.						
72-170-02-01	MRB	F72-99-99-000-808-F00	1.1	LIF LIM		ALL	ALL
	Discard right engine rear rotating (CDP) seal at manufacturer's life llimit. Note: Refer to the engine shop manual, chapter 5, for life limits.						
72-180-01-01	MRB	F72-00-00-200-802-F00 F72-00-00-200-805-F00 F72-00-00-200-814-F00	1.1 1.2	6600 FC 15000 FH	1600 FC 4000 FH	ALL	ALL
	Borescope inspection of the left engine combustion chamber. **ENGINE NOTE:** Task is for Single Annular Combustor engines.						
72-180-02-01	MRB	F72-00-00-200-802-F00 F72-00-00-200-805-F00 F72-00-00-200-814-F00	1.1 1.2	6600 FC 15000 FH	1600 FC 4000 FH	ALL	ALL
	Borescope inspection of the right engine combustion chamber. **ENGINE NOTE:** Task is for Single Annular Combustor engines.						
72-200-01-01	MRB	F72-00-00-200-802-F00 F72-00-00-200-806-F00 F72-00-00-200-814-F00	1.1 1.2	6600 FC 15000 FH	1600 FC 4000 FH	ALL	ALL
	Borescope inspection of the left engine HPT nozzle. (SAC /1 and DAC engines)						
72-200-01-02	MRB	F72-00-00-200-802-F00 F72-00-00-200-814-F00 F72-00-00-200-818-F00	1.1 1.2	6600 FC 15000 FH	1600 FC 4000 FH	ALL	ALL
	Borescope inspection of the left engine HPT nozzle. (Tech Insertion /3 and 7BE engines)						

INDEX

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
72-200-02-01	MRB	F72-00-00-200-802-F00 F72-00-00-200-806-F00 F72-00-00-200-814-F00	1.1 1.2	6600 FC 15000 FH	1600 FC 4000 FH	ALL	ALL
	Borescope inspection of the right engine HPT nozzle. (SAC /1 and DAC engines)						
72-200-02-02	MRB	F72-00-00-200-802-F00 F72-00-00-200-814-F00 F72-00-00-200-818-F00	1.1 1.2	6600 FC 15000 FH	1600 FC 4000 FH	ALL	ALL
	Borescope inspection of the right engine HPT nozzle. (Tech Insertion /3 and 7BE engines)						
72-210-01-01	MRB	F72-00-00-200-802-F00 F72-00-00-200-807-F00 F72-00-00-200-814-F00	1.1 1.2	6600 FC 15000 FH	1600 FC 4000 FH	ALL	ALL
	Borescope inspection of the left engine HPT blades.						
72-210-02-01	MRB	F72-00-00-200-802-F00 F72-00-00-200-807-F00 F72-00-00-200-814-F00	1.1 1.2	6600 FC 15000 FH	1600 FC 4000 FH	ALL	ALL
	Borescope inspection of the right engine HPT blades.						
72-220-01-01	MRB	F72-99-99-000-809-F00	1.1	LIF LIM		ALL	ALL
	Discard left engine HPT front shaft at manufacturer's life limit. Note: Refer to the engine shop manual, chapter 5, for life limits.						
72-220-02-01	MRB	F72-99-99-000-809-F00	1.1	LIF LIM		ALL	ALL
	Discard right engine HPT front shaft at manufacturer's life limit. Note: Refer to the engine shop manual, chapter 5, for life limits.						
72-230-01-01	MRB	F72-99-99-000-810-F00	1.1	LIF LIM		ALL	ALL
	Discard left HPT front rotating air seal at manufacturer's life limit. Note: Refer to the engine shop manual, chapter 5, for life limits.						
72-230-02-01	MRB	F72-99-99-000-810-F00	1.1	LIF LIM		ALL	ALL
	Discard right HPT front rotating air seal at manufacturer's life limit. Note: Refer to the engine shop manual, chapter 5, for life limits.						
72-240-01-01	MRB	F72-99-99-000-811-F00	1.1	LIF LIM		ALL	ALL
	Discard left engine HPT disk at manufacturer's life limit. Note: Refer to the engine shop manual, chapter 5, for life limits.						
72-240-02-01	MRB	F72-99-99-000-811-F00	1.1	LIF LIM		ALL	ALL
	Discard right engine HPT disk at manufacturer's life limit. Note: Refer to the engine shop manual, chapter 5, for life limits.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
72-250-01-01	MRB	F72-99-99-000-812-F00	1.1	LIF LIM		ALL	ALL
	Discard left engine HPT rear shaft at manufacturer's life limit. Note: Refer to the engine shop manual, chapter 5, for life limits.						
72-250-02-01	MRB	F72-99-99-000-812-F00	1.1	LIF LIM		ALL	ALL
	Discard right engine HPT rear shaft at manufacturer's life limit. Note: Refer to the engine shop manual, chapter 5, for life limits.						
72-270-01-01	MRB	F72-99-99-000-813-F00	1.1	LIF LIM		ALL	ALL
	Discard left engine LPT rotor support at manufacturer's life limit. Note: Refer to the engine shop manual, chapter 5, for life limits.						
72-270-02-01	MRB	F72-99-99-000-813-F00	1.1	LIF LIM		ALL	ALL
	Discard right engine LPT rotor support at manufacturer's life limit. Note: Refer to the engine shop manual, chapter 5, for life limits.						
72-280-01-01	MRB	F72-99-99-000-814-F00	1.1	LIF LIM		ALL	ALL
	Discard left engine LPT shaft at manufacturer's life limit. Note: Refer to the engine shop manual, chapter 5, for life limits.						
72-280-02-01	MRB	F72-99-99-000-814-F00	1.1	LIF LIM		ALL	ALL
	Discard right engine LPT shaft at manufacutrer's life limit. Note: Refer to the engine shop manual, chapter 5, for life limits.						
72-290-01-01	MRB	F72-99-99-000-815-F00	1.1	LIF LIM		ALL	ALL
	Discard left engine stage 1 LPT disk, stage 2 LPT disk, stage 3 LPT disk, stage 4 LPT disk at manufacutrer's life limit. Note: Refer to the engine shop manual, chapter 5, for life limits.						
72-290-02-01	MRB	F72-99-99-000-815-F00	1.1	LIF LIM		ALL	ALL
	Discard right engine stage 1 LPT disk, stage 2 LPT disk, stage 3 LPT disk, stage 4 LPT disk at manufacutrer's life limit. Note: Refer to the engine shop manual, chapter 5, for life limits.						
72-300-01-01	MRB	F72-56-00-200-802-F00	1.1	7500 FH	7500 FH	ALL	ALL
	Visual check of the left engine AFT mounts clevis for structural integrity failure.						
72-300-02-01	MRB	F72-56-00-200-802-F00	1.1	7500 FH	7500 FH	ALL	ALL
	Visual check of the right engine AFT mount clevis for structural integrity failure.						
72-320-01-01	MRB	F79-00-00-200-804-F00	1.1	500 FH	500 FH	ALL	ALL
	Detailed inspection of the left engine fwd sump, aft sump, AGB/TGB magnetic chip detectors or debris monitoring system detectors and scavenge screens for particles.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
72-320-02-01	MRB	F79-00-00-200-804-F00	1.1	500 FH	500 FH	ALL	ALL
	Detailed inspection of the right engine fwd sump, aft sump, AGB/TGB magnetic chip detectors or debris monitoring system detectors and scavenge screens for particles.						
73-010-01-01	MRB	F73-11-02-000-801-F00 F73-11-02-400-801-F00	1.1	6000 FH	6000 FH	ALL	ALL
	Replace the left engine fuel filter.						
73-010-02-01	MRB	F73-11-02-000-801-F00 F73-11-02-400-801-F00	1.1	6000 FH	6000 FH	ALL	ALL
	Replace the right engine fuel filter.						
73-020-01-01	MRB	F73-21-00-740-803-F00	1.1	150 FH	150 FH	ALL	ALL
	Interrogate the FMC CDU for left engine faults. **INTERVAL NOTE:** A. If any short time faults are found, corrective action for their repair is required immediately. The frequency of this check may be modified provided the new interval plus the time the fault corrective action may be deferred does not exceed 150 hrs total, as required per ATA 05-17-01 of the engine shop manual CFMI-TP.SM.10. For example, check recent faults every 70 hrs and fix the reported short time faults within the next 80 hrs. B. If any long time faults are found, corrective action for their repair is required with 425 hrs. The frequency of this check may be modified provided one half of the new interval plus the time the fault corrective action may be deferred does not exceed 500 hrs total, as required per ATA 05-17-01 of the engine shop manual CFMI-TP.SM.10. For example, check recent faults every 70 hrs and fix the reported long time faults within the next 465 flight hours. C. If any economic faults are found, repair is recommended on an opportunity basis.						
73-020-02-01	MRB	F73-21-00-740-803-F00	1.1	150 FH	150 FH	ALL	ALL
	Interrogate the FMC CDU for right engine faults. **INTERVAL NOTE:** A. If any short time faults are found, corrective action for their repair is required immediately. The frequency of this check may be modified provided the new interval plus the time the fault corrective action may be deferred does not exceed 150 hrs total, as required per ATA 05-17-01 of the engine shop manual CFMI-TP.SM.10. For example, check recent faults every 70 hrs and fix the reported short time faults within the next 80 hrs. B. If any long time faults are found, corrective action for their repair is required with 425 hrs. The frequency of this check may be modified provided one half of the new interval plus the time the fault corrective action may be deferred does not exceed 500 hrs total, as required per ATA 05-17-01 of the engine shop manual CFMI-TP.SM.10. For example, check recent faults every 70 hrs and fix the reported long time faults within the next 465 flight hours. C. If any economic faults are found, repair is recommended on an opportunity basis.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
73-030-01-01	CMR	F73-21-10-000-801-F00 F73-21-10-200-801-F00 F73-21-10-400-801-F00	1.1	300 FH	300 FH	800	ALL
	Remove the left engine hydro mechanical unit for inspection per Service Bulletin CFM 56-7B 73-016. **SPECIAL NOTE:** CMR task (73-CMR-01) interval for this task is 300 FH. See MPD Section 9. **ENGINE NOTE:** Applicable to engine hydro mechanical unit P/N 1853M56P04 or P/N 1853M56P05.						
73-030-02-01	CMR	F73-21-10-000-801-F00 F73-21-10-200-801-F00 F73-21-10-400-801-F00	1.1	300 FH	300 FH	800	ALL
	Remove the right engine hydro mechanical unit for inspection per Service Bulletin CFM 56-7B 73-016. **SPECIAL NOTE:** CMR task (73-CMR-01) interval for this task is 300 FH. See MPD Section 9. **ENGINE NOTE:** Applicable to engine hydro mechanical unit P/N 1853M56P04 or P/N 1853M56P05.						
74-020-01-01	MRB	F74-21-01-200-801-F00	1.1	4000 FC	4000 FC	ALL	ALL
	Detailed inspection of both left engine ignition leads.						
74-020-02-01	MRB	F74-21-01-200-801-F00	1.1	4000 FC	4000 FC	ALL	ALL
	Detailed inspection of both right engine ignition leads.						
78-011-01-01	MRB	F78-11-00-210-803-F00	1.1	ENG CNG		ALL	ALL
	Detailed inspection of the left engine exhaust plug drain pan and tube for condition and security. Note: This task is intended for on-aircraft use or if the short exhaust plug is still installed on the engine. Refer to CMM 78-11-40 for off-aircraft use or if the short exhaust plug has been removed from the engine. **AIRPLANE NOTE:** Applicable to airplanes with exhaust plugs equipped with drain pan and tube system installed. **ACCESS NOTE:** Engine exhaust plug removal required.						
78-011-02-01	MRB	F78-11-00-210-803-F00	1.1	ENG CNG		ALL	ALL
	Detailed inspection of the right engine exhaust plug drain pan and tube for condition and security. Note: This task is intended for on-aircraft use or if the short exhaust plug is still installed on the engine. Refer to CMM 78-11-40 for off-aircraft use or if the short exhaust plug has been removed from the engine. **AIRPLANE NOTE:** Applicable to airplanes with exhaust plugs equipped with drain pan and tube system installed. **ACCESS NOTE:** Engine exhaust plug removal required.						
78-050-01-01	MRB	F78-31-01-200-801-F00	1.1	15000 FH	15000 FH	ALL	ALL
	Visually check the left engine T/R's fan duct walls.						
78-050-02-01	MRB	F78-31-01-200-801-F00	1.1	15000 FH	15000 FH	ALL	ALL
	Visually check the right engine T/R's fan duct walls.						
78-060-01-01	MRB	F78-31-07-900-801-F00	1.1	12000 FH	12000 FH	ALL	ALL
	Detailed inspection of the left engine thrust reverser drag link spherical bearings.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
78-060-02-01	MRB	F78-31-07-900-801-F00	1.1	12000 FH	12000 FH	ALL	ALL
	Detailed inspection of the right engine thrust reverser drag link spherical bearings.						
78-070-01-01	MRB	F78-31-06-200-801-F00	1.1	12000 FH	12000 FH	ALL	ALL
	Visually check the left engine blocker doors.						
78-070-02-01	MRB	F78-31-06-200-801-F00	1.1	12000 FH	12000 FH	ALL	ALL
	Visually check the right engine blocker doors.						
78-080-01-01	MRB	F78-31-23-200-801-F00	1.1	15000 FH	15000 FH	ALL	ALL
	Visually check the left engine bullnose seal and retainer.						
78-080-02-01	MRB	F78-31-23-200-801-F00	1.1	15000 FH	15000 FH	ALL	ALL
	Visually check the right engine bullnose seal and retainer.						
78-100-01-01	MRB	F78-31-12-200-802-F00	1.1	7500 FH	7500 FH	ALL	ALL
	Detailed inspection of the left engine T/R fire seal.						
78-100-02-01	MRB	F78-31-12-200-802-F00	1.1	7500 FH	7500 FH	ALL	ALL
	Detailed inspection of the right engine T/R fire seal.						
78-110-01-01	MRB	F78-31-00-700-803-F00	1.1	5000 FH	5000 FH	ALL	ALL
	Perform an operational check of the left engine T/R sync lock. **SPECIAL NOTE:** CMR task (78-CMR-01) interval for this task is 5000 FH. See MPD Section 9.						
78-110-02-01	MRB	F78-31-00-700-803-F00	1.1	5000 FH	5000 FH	ALL	ALL
	Perform an operational check of the right engine T/R sync lock. **SPECIAL NOTE:** CMR task (78-CMR-01) interval for this task is 5000 FH. See MPD Section 9.						
78-120-01-01	MRB	F78-31-00-700-804-F00	1.1	3600 FH	3600 FH	ALL	ALL
	Perform operational check (bite) on the left engine EAU.						
78-120-02-01	MRB	F78-31-00-700-804-F00	1.1	3600 FH	3600 FH	ALL	ALL
	Perform operational check (bite) on the right engine EAU.						
78-130-01-01	MRB	F78-31-00-700-801-F00	1.1	15000 FH	15000 FH	ALL	ALL
	Perform an operational check of the left engine "reverser" light indication system.						
78-130-02-01	MRB	F78-31-00-700-801-F00	1.1	15000 FH	15000 FH	ALL	ALL
	Perform an operational check of the right engine "reverser" light indication system.						

TASK CARD NO.	SOURCE	AMM TASK REF	VERSION	INTERVAL THRESHOLD	INTERVAL REPEAT	APPLICABILITY AIRPLANE	APPLICABILITY ENGINE
79-010-01-01	MRB	F79-21-03-000-802-F00 F79-21-03-400-801-F00	1.1	7500 FH	7500 FH	ALL	ALL
	Remove and replace the left engine oil supply filter element.						
79-010-02-01	MRB	F79-21-03-000-802-F00 F79-21-03-400-801-F00	1.1	7500 FH	7500 FH	ALL	ALL
	Remove and replace the right engine oil supply filter element.						
79-040-01-01	MRB	F79-21-06-000-801-F00 F79-21-06-400-801-F00	1.1	7500 FH	7500 FH	ALL	ALL
	Remove and replace the left engine oil scavenge filter filter element.						
79-040-02-01	MRB	F79-21-06-000-801-F00 F79-21-06-400-801-F00	1.1	7500 FH	7500 FH	ALL	ALL
	Remove and replace the right engine oil scavenge filter filter element.						
80-010-01-01	MRB	F80-11-01-200-801-F00	1.1	1600 FC	1600 FC	ALL	ALL
	Detail inspection of the left engine starter magnetic chip detector for metal chips.						
80-010-02-01	MRB	F80-11-01-200-801-F00	1.1	1600 FC	1600 FC	ALL	ALL
	Detail inspection of the right engine starter magnetic chip detector for metal chips.						