

CHRISTIAN-ALBRECHTS-UNIVERSITÄT ZU KIEL

MASTER THESIS

Thesis Title

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Abstract

The Thesis Abstract is written here (and usually kept to just this page). The page is kept centered vertically so can expand into the blank space above the title too...

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Chapter 1

Motivation

Chapter 2

Pickup Ions

Pickup ions are created when neutral atoms inside the heliosphere become ionised and are subsequently swept away with the heliospheric magnetic field that is embedded within the solar wind.

2.1 The Heliosphere / Introduction

Oder Überkapitel Solar Physics?

Heliosphere: Grenze zu LISM

Solar Wind: Zusammensetzung, schneller und langsamer, high latitudes: less complex, constant in speed

B-Feldgleichung

Irgendwohin muss unbedingt Motivation, warum PUIs überhaupt interessant sind zu messen!

PU Process

2.2 Pickup Ions

A neutral atom inside the heliosphere is only subjected to the gravitational force and radiation pressure of the sun. It is not sensitive to any electromagnetic forces until it becomes ionised by solar ultra-violet radiation, charge exchange with solar wind protons or electron impact (Q?). After ionisation the particle starts interacting with the solar wind plasma. In particular it is forced onto a gyro orbit about the heliospheric magnetic field that is embedded within the solar wind. As the freshly created ion is swept away with the magnetic field line it is “picked up” from its location of ionisation – a new pickup ion (PUI) has been created.

PUIs were first observed by Möbius et al. (1985) with the SULEICA Instrument on the AMPTE spacecraft. The particles measured at 1 AU were He⁺ ions of interstellar origin.

Once the particle is ionised, its probability to become ionised another time decreases (Quelle). This characteristic of being only singly charged can help to discriminate PUIs from solar wind ions, that are mostly more often charged (Q?).

PUIs are mostly only single charged. This characteristic can help to distinguish them

from solar wind ions of coronal origin which often have been ionized multiple times, if not completely. (Q?)

VDF non-maxwellian, spatial density pattern

There have been observed several species of PUIs:

2.3 Interstellar Pickup Ions

Heliosheath, relative motion

The neutral part of the LISM can enter the heliosphere as it is not affected by the heliosheath (Todo). Inside the heliosphere the neutrals are guided only by the gravitational force and radiation pressure of the sun. The neutral particle's species determines how deep it can travel into the heliosphere before it becomes ionized. Species with a higher First Ionization Potential will be able to approach the sun much closer without being ionized. This results in He⁺ being the dominant PUI species at a solar distance of 1 AU even if in the LISM the abundance of hydrogen is about 10 times the one of helium.

- ionisation process is also dependent on the species
- radiation pressure only important for H (and He?). Kepler orbit...
- Spatial distribution:
gravitational force and radiation pressure lead to two regions of enhanced density of neutrals (in the ecliptic): Focusing cone and crescent. Focusing cone: For species with high FIP (as the others are ionized before and do not reach the downwind side of the sun)
- variation of He⁺ with the solar cycle: Rucinski 2003
- H, O and N are depleted in the filtration region (Baranov Malama 1995), Wimmer Skript: even before ionization: density is determined by ratio of gravitational force and photon pressure
- neutral density determines PUI production rate

2.4 Inner-source Pickup Ions

The idea of an additional source for the PUI's neutral seed population was born when Geiss, Gloeckler, Fisk, et al. (1995) measured a global distribution of C⁺ PUIs with the SWICS instrument on Ulysses. Interstellar carbon exists almost exclusively in a single charged state (Frisch et al., 2011) in the LISM. As only neutral atoms can enter the heliosphere it was not expected to find a distinct signature of C⁺ pickup ions. However, pickup carbon was observed with about the same ratio as oxygen, of which, in contrast, 80% is in a neutral charge state in the interstellar medium. These findings suggested that there must be another source for neutrals that has its origin somewhere inside the heliosphere.

In following studies (e.g. Geiss, Gloeckler, and Steiger, 1995) there were found also other species like O⁺ and Ne⁺ of these, so called, inner-source PUIs.

Inner-source PUIs show a composition that is similar to the one of the solar wind

(Gloeckler et al., 2000; Allegrini et al., 2005) as well as a velocity distribution function that is centered around $w_{SC} \approx 1$ (Schwadron et al., 2000) and seems to have thermalized with the solar wind.

Beneath those two characteristics there are other aspects concerning inner-source PUIs, that are still under debate. In particular that is the production mechanism of their neutral seed population. Allegrini et al. (2005) has summarized current candidates for possible scenarios. Two of those give an explanation for the ion's composition as they directly incorporate solar wind ions in the process:

- Solar wind recycling (Gloeckler et al., 2000; Schwadron et al., 2000): Absorption of solar wind ions by heliospheric grains and subsequent reemission of neutral atoms
- Solar wind neutralization (Wimmer-Schweingruber and Bochsler, 2002): Solar wind ions penetrate sub-micron-sized dust grains and undergo (partial) neutralization by charge exchange

(As this work does not focus on inner-source PUIs in particular...)

2.5 VDF

After the particle has been ionised it is forced onto a gyro motion about the local field line of the heliospheric magnetic field due to the Lorentz force.

To examine the velocity distribution of PUIs after they have been ionized we need to consider the initial speed v_{ini} of the neutral particle. For neutrals from the LISM this is mainly given by the inflow speed v_{ISM} of the local interstellar medium with which they enter the heliosphere. As we don't exactly know about the production mechanism of inner-source PUIs, the following considerations mainly relate to interstellar PUIs. Schwadron et al. (2015) obtained $v_{ISM} \approx 25 \text{ km s}^{-1}$ with the IBEX satellite for helium. Considering the acceleration by the sun's gravitational force we have a maximum initial speed of $v_{ini} \approx 50 \text{ km s}^{-1}$ at 1 AU. Compared to an average solar wind speed of $v_{sw} \approx 400 \text{ km s}^{-1}$ one can neglect this initial speed in a first step.

For simplicity we thus consider a neutral particle at rest that becomes ionized by one of the aforementioned processes. The freshly created ion now is subjected to the electromagnetic forces of the solar wind plasma. In particular, it finds itself at a velocity v_{sw} relative to the magnetic field which is convected outwards by the solar wind that is assumed to flow radially outwards. Due to the Lorentz force the PUI starts to gyrate about the magnetic field line on an orbit that is perpendicular to it. When we further consider a magnetic field's orientation that is perpendicular to the solar wind flow, $\vec{B} \perp \vec{v}_{sw}$, the ion's gyration speed is v_{sw} while its guiding center moves together with the field line at a speed of v_{sw} as well. Thus, the total speed of the PUI ranges between $0 v_{sw}$ and $2 v_{sw}$ in a sun frame of reference.

As the heliospheric magnetic field lines are shaped like an Archimedean spiral, the so called *Parker spiral*, the assumption of a perpendicular magnetic field only applies when solar wind speed v_{sw} and solar distance r_{\odot} follow the relation

$$90^\circ \approx \arctan \left(\frac{2\pi}{T_{\odot} \cdot v_{sw}} r_{\odot} \right)$$

with sun's sidereal period $T_{\odot} \approx 25 \text{ d}$ (Prölss, 2004). In other cases, e.g. for solar distances about 1 AU, at which the angle between solar wind and magnet field direction

is approximately 45° , the maximum speed in a sun frame of reference is decreased. In general, the gyration speed is given by

todo

with

The velocity space for a pickup situation with a non-radial magnetic field orientation is shown in figure *todo* on the right. The PUI's total velocity consists of the movement of the guiding center (...) and the gyration velocity (...). We note, that in this case there is a relative velocity between the motion of the solar wind bulk and the PUI's guiding center movement.

However, independent on the magnetic field orientation, every possible velocity space trajectory is part of a sphere with the radius v_{sw} centered around \vec{v}_{sw} . That means that, in the frame of the solar wind, the freshly created PUI always moves with a speed that is as fast as the solar wind itself. (*todo: Hier w einführen?*)

Instead of a single PUI we can consider an ensemble of PUI's that is injected into the solar wind while the magnetic field orientation is not changing much. For that we expect the VDF to form a ring shape in velocity space, commonly called the "PUI torus VDF" (Oka et al., 2002). The expected orientation of this highly anisotropic torus VDF depends on the local magnetic field direction and is sketched in figure *todo* for three different angles.

...thickness that is related (associated) to the neutral's velocity and is very small compared to the radius...

Spatial diffusion: chalov Fahr 1998. Signature of plasma parcel in which it was produced doesn't match with the one it is measured in

After the injection, the PUI population is radially carried away with the solar wind. During phase space transport through the heliosphere the PUIs are subjected to multiple processes that are expected to modify the shape of the initial toroidal VDF. However, it is not completely understood how the VDF evolves in detail.

A fast isotropization of the VDF due to pitch-angle scattering was suggested by Vasyliunas and Siscoe (1976) in a theoretical work. However, observations by e.g. Möbius et al. (1998) on *He+* or Gloeckler et al. (1995) on TODO have shown clear anisotropic features in the measured VDFs. Following studies (*todo*) explained these findings with the assumption that the ions would be injected into the sunward hemisphere of velocity space more likely. Ineffective pitch-angle scattering into the anti-sunward hemisphere thus would result in an radial anisotropy.

Recent observations have emphasized the influence of the magnetic field direction on the measured anisotropy. Utilising 2D analyses of the velocity space, Oka et al. (2002) and Drews et al. (2015) found that the measured VDF of PUIs is systematically oriented about the direction that is perpendicular to the magnetic field. Thus, it is believed that the VDF's anisotropic features are remnants of the initial toroidal VDF. (and the pa scattering didn't have enough time to isotropize the distribution)

Furthermore, there are different acceleration and deceleration processes that change

the PUI's initial VDF and lead to a diffusion in velocity space. Under the assumption of an isotropic VDF the PUI population is often treated as an adiabatic gas that is consequently cooled when expanding with the solar wind. This picture, initially suggested by Vasyliunas and Siscoe (1976), however, must be reviewed due to the doubtful fact of a fully isotropic VDF. Another cooling mechanism, called the *magnetic cooling*, is due to the magnetic field weakening with solar distance. As the PUIs are swept outwards both their ... and their ... invariant have to be conserved which leads to a decrease in both velocity components (parallel and perpendicular to the magnetic field) and thus to a decrease in total velocity (in the frame of the solar wind).

(focusing (adiabatic invariant) & Ginzberg Landau (Fahr2008): "magnetic cooling"
(auch gute Erklärung: Fahr&Fichtner2011))

Acceleration of PUIs can be caused by acceleration: first and second order fermi (verstehen, Gründe): eher außen bzw. eher innen. Außerdem ein Mechanismus, der nicht an einzelne Events gebunden ist, sondern immer vorhanden: Mechanismus für alle Teilchen, power law -5...

man kann in der 1D Verteilung beobachten, dass $2v_{sw}$ exceeded wird

PUI He should be measured throughout the mission as they penetrate the heliosphere until 0.5 AU (Gloeckler et al., 1992)

Instrument that is capable of measuring this distribution: large acceptance in absolute velocity, large variation and resolution in angles

2.5.1 1D reduced VDF, aim of this work...?

Warum sw frame?

Warum brauchen wir 3D? Um das zu entfalten und dann den Frame wechseln zu können.

Chapter 3

Instrumentation

3.1 Ulysses

The Ulysses spacecraft (Wenzel et al., 1992) was launched in 1990 and orbited the sun for nearly 20 years as a joint ESA/NASA project. Ulysses' most remarkable feature is its out-of-ecliptic orbit with a maximum heliographic latitude of 80.1° . As the first spacecraft it was hence capable of taking in situ measurements from above the poles of the sun.

The primary goal of the mission was to study the heliosphere in three dimensions. Some of the original main objectives were:

- to study the interplanetary magnetic field and the solar wind, especially its composition, the origin and waves and shocks within the solar wind plasma
- to investigate galactic cosmic rays and energetic particles
- to improve the knowledge about interplanetary dust
- to explore the neutral component of interstellar gas

Some secondary objectives included e.g. the investigation of Jupiter's magnetosphere during the Jupiter flyby and the search for gravitational waves and for gamma-ray burst sources (Wenzel et al., 1992).

For these aims Ulysses was equipped with a wide range of different instruments and antennas. One of the in situ instruments is the Solar Wind Ion Composition Spectrometer (SWICS), that will be described in the next section.

A sketch of Ulysses' unique orbit is shown in figure 3.1. Ulysses was launched in October 1990 and left earth's gravitational field with 15.4 km/s. Starting with a flyby manoeuvre around Jupiter Ulysses was sent onto its highly elliptical orbit. With an orbital period of 6.2 years Ulysses completed nearly three orbits around the sun until communication was shut down in June 2009 due to the expiring of the radioisotope thermal generators. Within the mission's long lifetime the Sun's behaviour over its activity cycle of 22 years could be studied.

Ulysses is a spin-stabilized spacecraft that spun at 5 rpm. The spin axis is aligned with the high-gain antenna's electrical axis, that provided a communication link from the spacecraft to Earth. The downlink bitrate was variable with up to 1024 bit/s during real-time connection.

TODO: Importance PUIs?

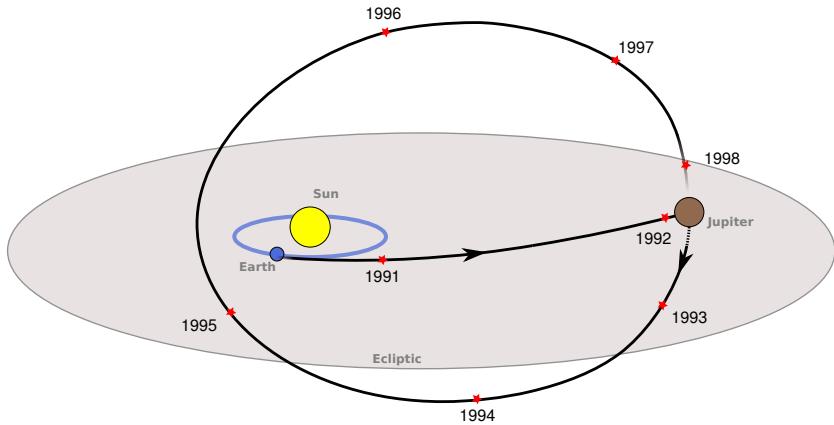


FIGURE 3.1: A sketch of Ulysses' first orbit. After the launch from earth in 1990 the spacecraft was sent to Jupiter from where it left the ecliptic on an elliptical orbit around the sun with perihelion at 1.3 AU and aphelion at 5.4 AU. Due to the orbit's high latitude of $\sim 80^\circ$ Ulysses crossed the Sun's pole regions two times between 1994 and 1996. Figure after European Space Agency (2019).

3.2 SWICS

The Solar Wind Ion Composition Spectrometer (SWICS, Gloeckler et al. (1992)) is a time-of-flight mass spectrometer mounted on the spacecraft Ulysses (s. sec. 3.1). The instrument is designed to determine the elemental and charge-state composition and the velocity distribution of solar wind ions. With an energy-per-charge range from 0.16 kV to 59.6 kV SWICS is able to measure every solar wind ion species from protons to iron with any typical charge state. Depending on the individual ion, particle energies from 1 keV up to 1 MeV are covered.

SWICS is mounted on the sun-facing side of Ulysses and revolves around the spacecraft's spin axis with the spacecraft spinning. A photograph of the instrument is shown in fig. 3.2, left.

Ulysses SWICS also has a twin instrument that is the SWICS spare model, which has been mounted on the ACE spacecraft (Stone et al., 1998).

TODO: Super für measuring PUIs! viele wichtige Messungen...

SWICS measures the mass m , the charge q and the energy E of entering ions by a combination of three separate measurements: The electrostatic deflection analyzer within the entrance systems is used for determining the energy-per-charge of a particle. Within the time-of-flight/energy section the particle's time-of-flight and residual energy are measured. A more detailed description of the measurement is given in the next sections. A particle's trajectory through the instrument can be tracked in the schematic in fig. 3.3.

3.2.1 Collimator and Electrostatic Analyzer

Particles enter the instrument through the entrance collimator. It restricts particles to the ones with a trajectory that is parallel to the collimator slits. The collimator follows an intricate geometry that is fan-shaped with an opening angle of 69° in width and 4° in height and that is at the same time curved along its width. Fig. 3.2,



FIGURE 3.2:

Left: Photograph of the SWICS instrument. The fan-shaped collimator on top of the electrostatic deflection analyzer can be seen on the right. It is followed by the cylindrically shaped and silvery coloured *high-voltage bubble*, that contains the time-of-flight system, analog electronics and the sensor power supply. The gold-plated cylinder houses a -30 kV voltage supply. The opening angles of the collimator are 69° in width and 4° in height. OPT: Ausschneiden und beschriften

Right: Schematic cut through the sensor. In this view the orientation of the three solid state detectors is visualized. OPT: farbig markieren

Both figures are from from Gloeckler et al. (1992).

left, gives an idea of the shape.

After having entered through the collimator a particle has to pass the electrostatic analyzer. It is split up into two sections – energies-per-charge from 0.16 to 14 kV are covered by the proton/helium channel. Particles within this range of energy will be filtered by their energy-per-charge and after an post-acceleration will be counted by a solid-state detector. As this simple measurement principle is very limiting for our analysis we will focus on the main channel that is suitable for a full *mass/mass-per-charge* analysis.

The main channel covers an energy-per-charge range of 0.66 to 60.51 kV . A particle can only pass through the pair of curved deflection plates if its kinetic energy-per-charge equals a certain ratio that is given by the voltage between the two plates. To measure particles of different energy-per-charge the deflection voltage is stepped through 64 logarithmically spaced values, s. fig 3.4. As the voltage steps once per spin of Ulysses (every 12 s), a complete voltage cycle lasts ~ 12.8 minutes. Every step has a relative uncertainty in energy-per-charge of $\Delta\text{EpQ}/\text{EpQ} = 5\%$ that is due to the finite space between the plates.

3.2.2 Time-of-flight measurement

After having passed the electrostatic analyzer the ion is post-accelerated by an constant potential drop of 30 kV . It enters the time-of-flight chamber with penetrating through a thin ($\sim 3\text{ }\mu\text{g/cm}^2$) carbon foil from which secondary electrons are emitted. These electrons are guided to a microchannel plate detector where a start signal is triggered (s. fig 3.3). The stop signal is triggered after the ion has traversed nearly force free a distance of 10 cm through the vacuumized time-of-flight section. Here it hits one of the solid-state detectors, where secondary electrons are emitted again. By

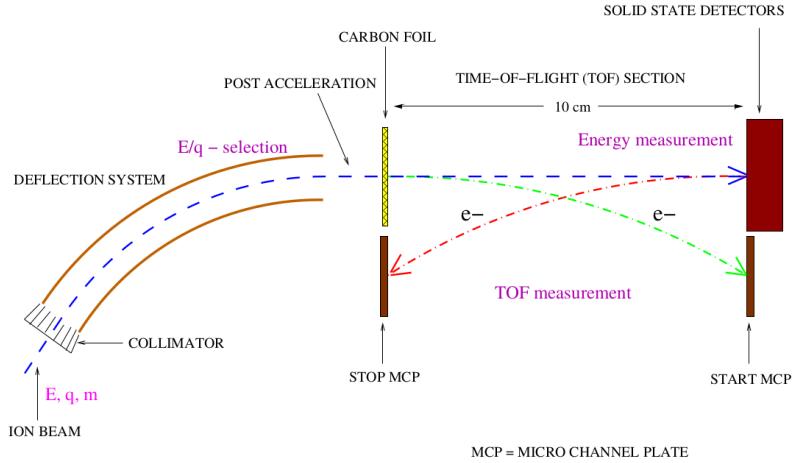


FIGURE 3.3: Schematic view of the SWICS detector. See text for reference. This figure is taken from Berger (2008).

combining the times of the start and the stop signal the particle's time-of-flight has been measured.

3.2.3 Energy measurement

Additionally, the ion's energy is measured when they hit one of the three solid state detectors.

A solid-state detector is realized by applying an inverse bias voltage to a semiconductor material. When ionized particles hit the material, they set free charge carriers from the resulting depletion zone. These charge carriers can be measured as a pulse of current which is proportional to the deposited energy of the ion.

Each of the three detectors has an active area of $1.5 \times 1.3 \text{ cm}^2$. Their alignment with each other can be seen in fig 3.2, right: While the central detector is oriented perpendicular to the symmetry axis of SWICS, the other two detectors are slightly tilted with respect to this axis. This way particles with different angles of incidence along the width of the collimator can be detected. The information on which of the three detectors has been hit will be utilized in chapter 4 for a coarse directional resolution of incident particles.

One of SWICS' main objectives is the measurement of the composition of incident particles. An ion and its charge state can be fully identified with the knowledge of its mass m and its mass-per-charge mpq . With the measurement of the time-of-flight ToF , the energy E_{SSD} and the knowledge of the energy-per-charge EpQ we can calculate the mass m , the mass-per-charge mpq and velocity v of an ion i with the following set of equations:

$$m_i = 2 E_{SSD} \left(\frac{Tof}{d} \right)^2 \quad (3.1)$$

$$mpq_i = 2 (EpQ + V_{PAC}) \cdot \left(\frac{Tof}{d} \right)^2 \quad (3.2)$$

$$v_i = \sqrt{2 EpQ \frac{1}{mpq_i}} \quad (3.3)$$

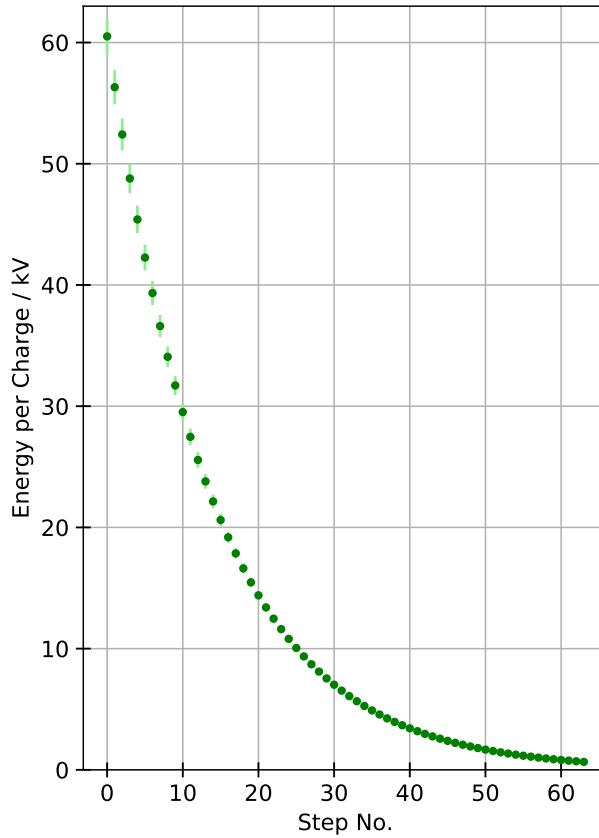


FIGURE 3.4: Energy-per-Charge steps of Ulysses SWICS. A voltage cycle starts with step 0 that is the highest voltage $E_pQ = 60.51$ kV. Voltage is then swept down 64 logarithmically spaced steps up to $E_pQ = 0.66$ kV. The uncertainty of the single steps is $\frac{\Delta E_pQ}{E_pQ} = 5\%$ and drawn in lighter color.

where d is the length of the time-of-flight section and V_{PAC} is the post-acceleration voltage. v denotes the ion's initial velocity when entering the instrument and is not to be confused with its velocity during the time-of-flight measurement, that is altered particularly by the post-acceleration.

3.3 Data products

Direct Pulse-Height Analysis data (PHA data) are one of SWICS' data products and the ones that are most relevant for the analysis in this work.

Over one voltage cycle SWICS steps through the 64 energy-per-charge steps of the electrostatic analyzer (s. sec 3.2.1), which we call ESA steps. At normal spacecraft telemetry rates SWICS steps once per spin, which is once every 12 s. During these times 30 PHA words per spin are selected and transmitted. The selection is based on a priority scheme that is described below in sec. 3.3.2.

Every 24-bit PHA word contains following information on an incident particle that triggered a valid measurement:

- E_{SSD} : Energy deposition in the solid state detector measured through 256 channels in a range 40 – 600 keV

- *ToF*: Time-of-Flight
measured through 1024 channels in a range 10 – 200 ns
- Sector information:
SWICS divides one spin of Ulysses up into 8 sectors of approximately equal duration. This yields information on the spatial origin of particles.
- Detector information:
Which of the three solid state detector elements has been triggered?
- Priority category:
Due to limited telemetry only an assorted sample of all measured particles can be transmitted. This selection is based on different priorities. For details see sec [3.3.2](#).

The interpretation of a set of PHA words that have been collected over time is discussed in sec. [3.4](#).

3.3.1 Detection Efficiency

When working with SWICS – like for any physical measurement – one has to consider several constraints that impede the ideal measurement as it is described in sec. [3.2](#).

One of these constraints is the detector efficiency, which describes the probability to measure a particle that has entered the instrument (Todo: ?). With an ion passing through the time-of-flight section there is the likelihood that secondary electrons may not be emitted properly from the carbon foil and solid state detector which then leads to an invalid time-of-flight measurement. Also, ions could pass through the time-of-flight section on divergent trajectories due to scattering processes (TODO: in the carbon foil??). Subsequently, the ion possibly does not hit the sensitive area of the solid state detector and would neither trigger a stop signal for the time-of-flight measurement nor a valid energy measurement in the solid-state detector.

Another reason for an invalid energy measurement in the solid-state detector can be that the ion's energy is smaller than the threshold of the solid-state detector. In this case, only energy-per-charge and time-of-flight information for the ion are available. Such events without a corresponding energy measurement are called double coincidences, while events with triggered start and stop signal and an energy measurement are called triple coincidences.

The reason for choosing a non-zero threshold is to limit the influence of the solid-state detector's natural noise. For SWICS this noise level is quite high, at around 12 keV (Gloeckler et al., [1992](#)) (todo: ?), so that the threshold is chosen to be ~ 30 keV. The detection efficiency is highly dependent on the ion species and deflection step. Ions with a small mass and charge are most likely to not overcome the threshold at low energy-per-charge values. For the bulk of He^+ this is already the case for ESA step 17 ($E_p Q = 17.86$ kV), which corresponds to a He^+ velocity of $v = 900 \text{ km s}^{-1}$ (s. eq. [3.3](#)).

Unfortunately, He^+ detection efficiencies are not known to us for Ulysses SWICS. Instead, we make use of the efficiencies from ACE SWICS, that have been calculated by Kötén ([2009](#)). As ACE SWICS uses different energy-per-charge ranges we had to extrapolate the values for Ulysses SWICS. For the highest energy-per-charge value $E_p Q = 60.51$ keV we interpolated the efficiency from the ACE He^+ triple efficiencies and then extrapolated linearly to an efficiency of 0 at ESA step 17. By this, we accommodate for the above mentioned fact that He^+ does not have enough energy

to deposit energy above the solid state detector's threshold at ESA step 17 and thus, the probability for a triple coincidence is zero. He^+ efficiencies from ACE SWICS and the resulting efficiencies used for Ulysses SWICS are shown in fig. 3.5. For sure, these values are not realistic but represent the overall trend of a decreasing efficiency for higher ESA steps.

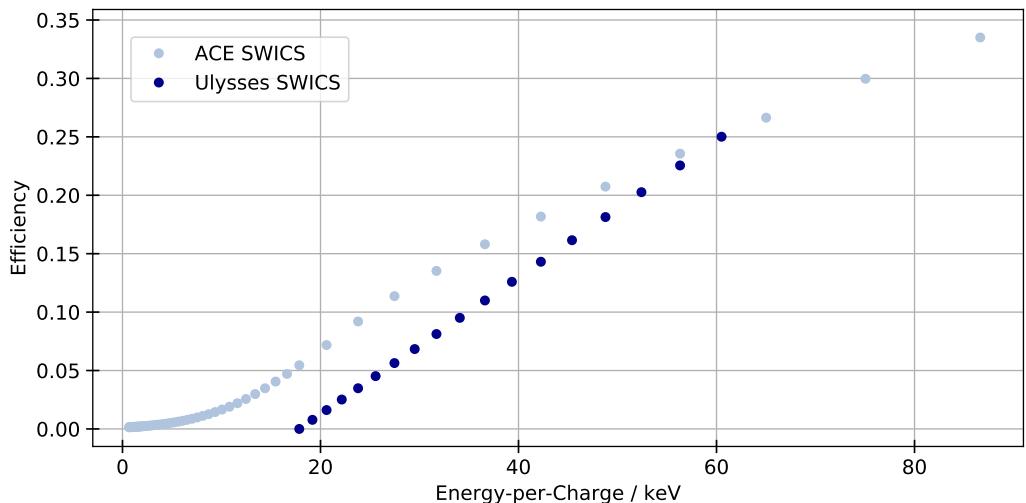


FIGURE 3.5: He^+ detection efficiencies for ACE SWICS over all ESA steps from 0.66 kV to 86.64 kV. For Ulysses SWICS the efficiencies have been extrapolated between 0 for ESA step 17 ($E_{pQ} = 17.86 \text{ kV}$) and the highest efficiency for ESA step 0 ($E_{pQ} = 60.51 \text{ kV}$). The latter value has been interpolated from the ACE efficiencies.

3.3.2 Priority Weighting

SWICS' data processing unit performs an on-board mass and mass-per-charge-classification. For every ion with valid measurements the mass m and mass-per-charge mpq are calculated based on the measurements of E_{SSD} , ToF and the particle's ESA step using a look-up-table technique.

On the one hand these values are used for sorting the ions into predefined boxes in the m - mpq -space which yield in the so-called matrix rate, a second data product that is provided beneath the PHA words.

Secondly, and more important for our analysis, the m and mpq information is used to sort every measured ion into one of three priority ranges. Due to a limited telemetry possibly not every measured particle can be transmitted as a PHA word. 30 PHA words per spin are chosen from these ranges in way that the ratio of low and high priority PHA words is balanced. By this selection less abundant heavier ions are artificially enhanced which is necessary to recreate the measured composition from the transmitted PHA words. Table 3.1 shows a summary of this priority scheme. Details about this can be found in Gloeckler et al. (1992).

To account for the priority biased selection of PHA words it is necessary to weight the transmitted data with the so-called base-rate weight $brw = \frac{D_{\text{PHA}}}{T_{\text{PHA}}}$, where T_{PHA} is the number transmitted PHA words and D_{PHA} the number of detected particles on which this PHA word is based. The measured composition can be restored in this way.

TABLE 3.1: SWICS' priority categorization scheme

Range 0	$m < 8.7$; $m_0 : mpq < 3.3$	H, He ⁺ , He ²⁺ ; Doubles
Range 1	$m > 8.7$	Heavier ions
Range 2	$m_0 : mpq > 3.3$	Doubles

3.4 ET matrices, identification

Fig. 3.6 shows longterm PHA data collected over two years with Ulysses SWICS for ESA step 24 ($EpQ = 10.8$ kV). For every particle with valid time-of-flight and energy measurements the E_{SSD} channel is plotted over the time-of-flight channel. As a particle's mass and mass-per-charge are connected to the measured values EpQ , ToF and E_{SSD} by equations 3.1 – 3.3, every ion species occupies a distinct position in this histogram, which we call ET-matrix. However, due to uncertainties in the measurements of EpQ , ToF and E_{SSD} an event may be displaced around its ideal position. For multiple events of the same species this results in broadened distributions. A detailed discussion on these uncertainties can be found for example in Berger (2008). We can identify several ion species in the shown ET-matrix: A selection of those is labelled in 3.6.

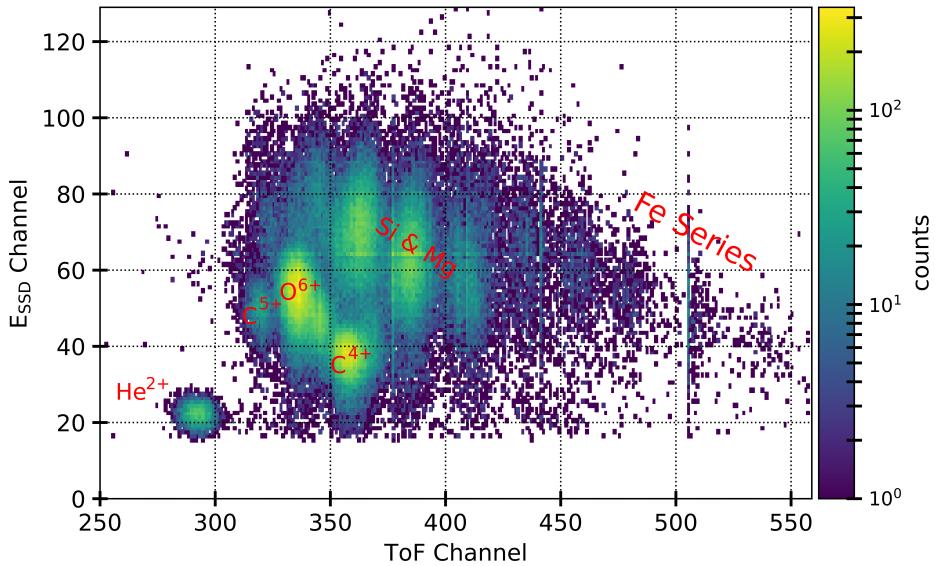


FIGURE 3.6: ET-matrix from Ulysses SWICS PHA data of years 1993–1994. All plotted events were measured during ESA step 24 ($EpQ = 10.8$ kV). Only triple coincidences are shown. Besides accumulated counts in ion-specific positions, there are a few horizontal and vertical stripes apparent in the shown ET-matrix – for example in E_{SSD} channel 65 and time-of-flight channel 409, 377 or 505. These are imprints of irregularities within SWICS' Analog-Digital-Converters which can be noticed as a saturation of one bin while the adjacent bin is depleted. However, these signatures are not expected to have a significant effect on the following analysis.

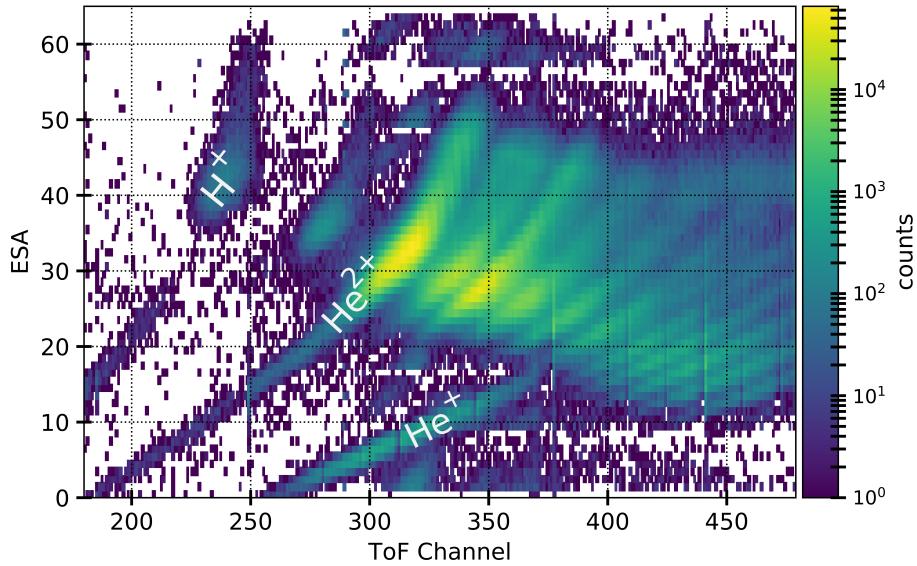


FIGURE 3.7: Histogram over time-of-flight and ESA channels of two years (1993-1994) of Ulysses SWICS Triple Coincidence PHA data. Note, that the ESA channel numbers are inversely proportional to the energy-per-charge values. Ion species with the same mass-per-charge ratio can be detected on curves $ESA \sim ToF^2$.

3.4.1 Selection of He^+

For extracting only the He^+ events from the PHA data we histogram the events' energy-per-charge against their measured time-of-flight in fig. 3.7. Ions with the same mass-per-charge can be tracked on curves with $ESA \sim ToF^2$, while species with higher masses-per-charge can be found on curves at higher ToF channels. This way one can identify for example H^+ and He^{2+} .

At the bottom of fig. 3.7 suprathermal He^+ clearly stands out. For time-of-flight channels above ~ 375 heavier solar wind ions with similar mass-per-charge ratios (e.g. Si^{7+}) mix into the He^+ population. We can take advantage of SWICS' on-board priority weighting, that has been described in sec. 3.3.2. By selecting Triple Coincidences with Range-0 priority only we separate out events with $m > 8.7 \text{ u}$. This leaves us mainly with H^+ , He^{2+} and He^+ as the prominent ions in the Range-0 mass range. The result is shown in fig. 3.8. One can also see traces of heavier ions seeping into the Range-0 regime but this does not affect those ESA steps in which He^+ is present. For a final selection we define a box around the He^+ population and extract all events within this box. The result is shown in fig. 3.9, above.

As explained in sec. 3.3.1, with He^+ we are limited to ESA steps 0 to 19 when only considering Triple Coincidences. For higher ESA steps He^+ only occurs as a double coincidence. Double coincidences are not sufficient for us, as we need the information about which solid-state detector has been triggered for a directional analysis.

The presented selection procedure has been performed on the complete data set from 1991 to 2009 resulting in an average of $\sim 6.8 \cdot 10^6 \text{ He}^+$ PHA events per year. This is $\sim 165 \text{ He}^+$ events per voltage cycle. Note, that there is a wide variation between different periods of time. PHA numbers per year are listed in fig. 4.12.

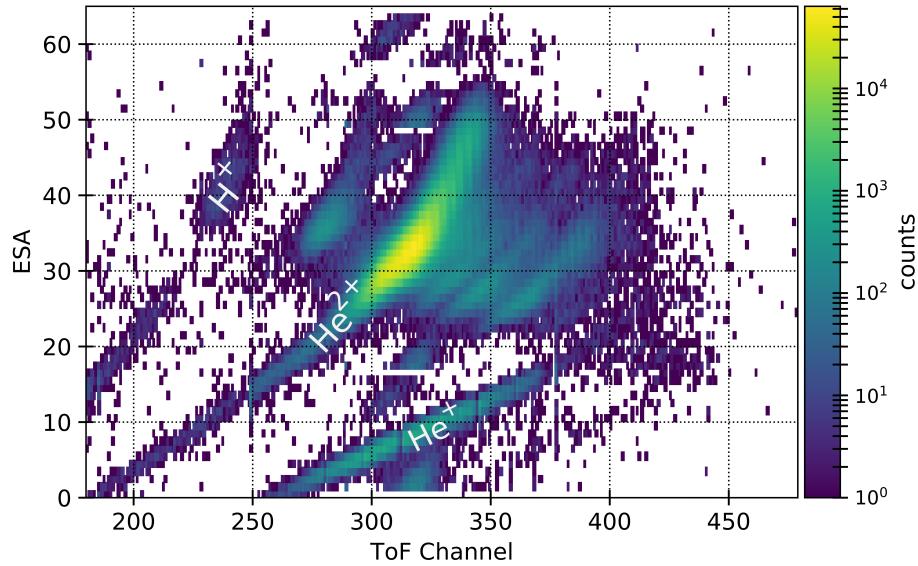


FIGURE 3.8: The same histogram as in fig. 3.7, but this time only events with Range-0 priority are plotted. This basically removes ions with $m > 8.7 \text{ u}$.

3.4.2 Selection of He^{2+}

For testing the virtual detector in chapter 4 we need a species of which we know that it mainly moves with the solar wind bulk and that occurs within the Triple Coincidences, so that we have directional information. We choose for He^{2+} , the most abundant ion in the solar wind (Prölss, 2004, ch. 6.1) as it also occurs in Range 0 and is easily recognizable in fig. 3.8. Unlike He^+ , He^{2+} can also be found at ESA steps higher than step 19, as it gains more energy when being post-accelerated due to the twice as high charge. Thus, He^{2+} deposits more likely energies above the threshold of the solid-state detector. We selected He^{2+} with the same procedure as for He^+ . However, in this case it is not so crucial to include exactly every He^{2+} event, as we do not want to perform a full analysis with He^{2+} . The resulting selection is shown in 3.9, below.

TODO: woher sind überhaupt die PHA Daten?

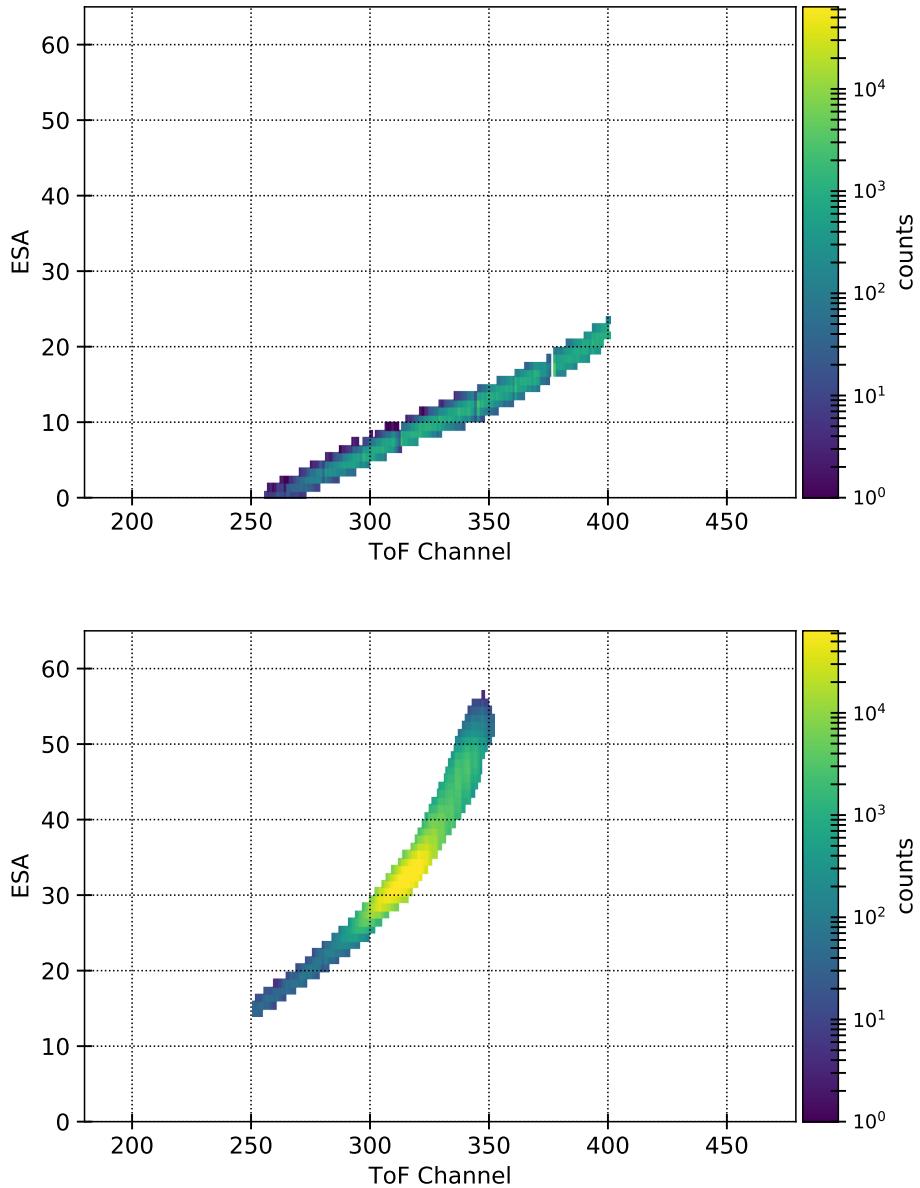


FIGURE 3.9: He^+ (above) and He^{2+} (below) selections that have been cut out from fig. 3.8.

Chapter 4

Data Analysis and Methods

In this chapter we combine the prepared SWICS data from chapter 3 with position and orientation information of the Ulysses spacecraft with help of a model of the SWICS detector.

Todo: (This way we can take the step from the directional incident information of a measured ion to its position in velocity space.)

The virtual detector is modeled after the true geometry of the SWICS instrument. By taking into account

In a first step the two main coordinate systems of this work are introduced.

4.1 Coordinate Systems

In the following analysis the SWICS PHA data will be connected to the position and orientation of the Ulysses spacecraft. For this two different coordinate systems are utilized.

4.1.1 Heliographic Coordinate System

Trajectory data from Ulysses Final Archive (2008) is used in this work to describe Ulysses' orbit. In this data, Ulysses' position is given in daily intervals and based on different coordinate systems, including the heliographic coordinate system in the B1950.0 epoch.

The heliographic (HG) coordinate system is a Cartesian Sun-centered system with the Sun's equatorial plane as a reference plane. Its X-axis is directed along the line of ascending node, which is the intersection line of the ecliptic and the solar equatorial plane. While the latter has an inclination of $i_{\odot} = 7.25^\circ$ against the ecliptic (Fränz and Harper, 2002) the line of ascending node is at an ecliptic longitude of $\Omega_{\odot} \approx 75^\circ$ relative to the First Point of Aries in 1950 (NASA HelioWeb, 2019). The Z-axis of the HG coordinate system is directed along the Sun's spin axis (northward) and the Y-axis completes the right-handed system.

In HG spherical coordinates the longitude φ_{HG} is defined to be 0° for directions along the X-axis and increases towards the Y-axis. The latitude ϑ_{HG} is 0° for directions within the solar equatorial plane and $+90^\circ$ for northward directions.

However, when working with the Ulysses trajectory data it was found that these data were given in spherical coordinates for which $\varphi_{\text{HG}} = 0^\circ$ was towards -105° ecliptic longitude relative to the First Point of Aries. This means that the Ulysses trajectory coordinate system is shifted 180° around the solar spin axis against the classical definition (s.a.).

In fig. 4.1 Ulysses' spherical HG coordinates as well its distance from the Sun are given over time of the mission.

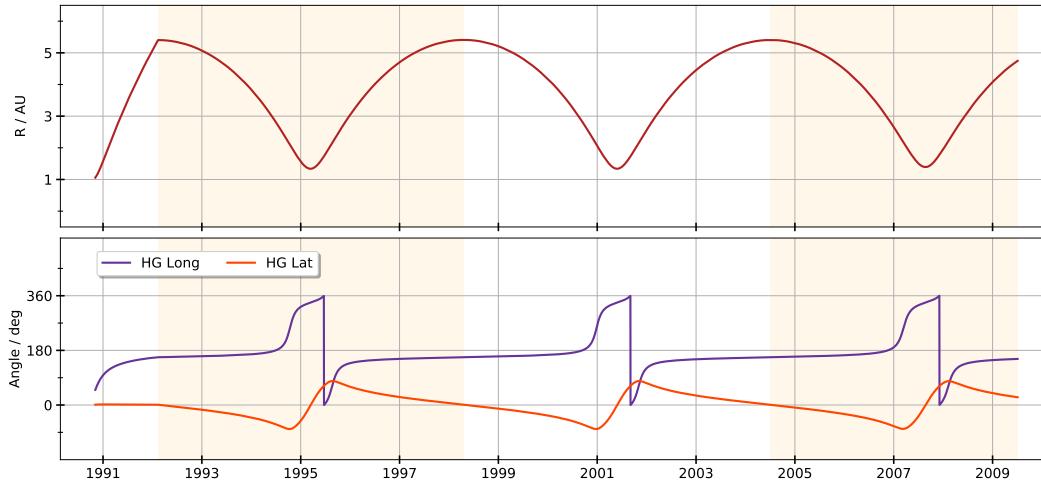


FIGURE 4.1: Ulysses trajectory data from November 1990 to June 2009 based on the data from the Ulysses Final Archive (2008). Color shaded are the three orbits of the mission. **Above:** Shown is Ulysses' radial distance R from the Sun. Perihelion and aphelion of the space-crafts' elliptical orbit can clearly be seen as the nearest/farthest distance that is 1.3 AU and 5.4 AU. Over the duration of the mission Ulysses passes each of these points every 6.2 years and three times in total. **Below:** Shown are Ulysses' HG longitude and latitude. Note, that the longitude is shifted around 180° with respect to the classical definition (s. text for details). A clear periodicity over Ulysses' three orbits can be seen. Maximum and minimum latitude are $\sim \pm 80^\circ$. These points are highest above the poles of the Sun and are reached a few weeks before and after Ulysses' pass through the perihelion.

TODO: Fontsize größer

4.1.2 Radial Tangential Normal Coordinate System

Ulysses' orbit has been described in the HG coordinate system, which is a coordinate system that is fixed with respect to the Sun. For describing positions and velocities in the frame of the spacecraft (e.g. the Aspect Angle) we need a coordinate system that moves with the spacecraft.

When dealing with Ulysses' trajectory data it is useful to work with Radial Tangential Normal (RTN) coordinates. The RTN coordinate system is defined relative to a moving object in the heliosphere, in this case Ulysses, and is centered at the Sun. A graphical representation of the system is given in fig. 4.2. The unit vectors are \vec{R} , \vec{T} and \vec{N} , where \vec{R} points radially outward from the Sun to the current position of the spacecraft. \vec{T} is defined as the normalized cross product of the Sun's angular velocity, $\vec{\omega}$, and \vec{R} . \vec{N} completes the right-handed Cartesian coordinate system. Consequently, the RTN system is not defined for a spacecraft's position right above one of the Sun's poles as the cross product $\vec{\omega} \times \vec{R}$ is zero here. Nevertheless we do not have to worry about this fact as not even Ulysses crosses the poles directly.

OPT: Umrechnung von HG in RTN?

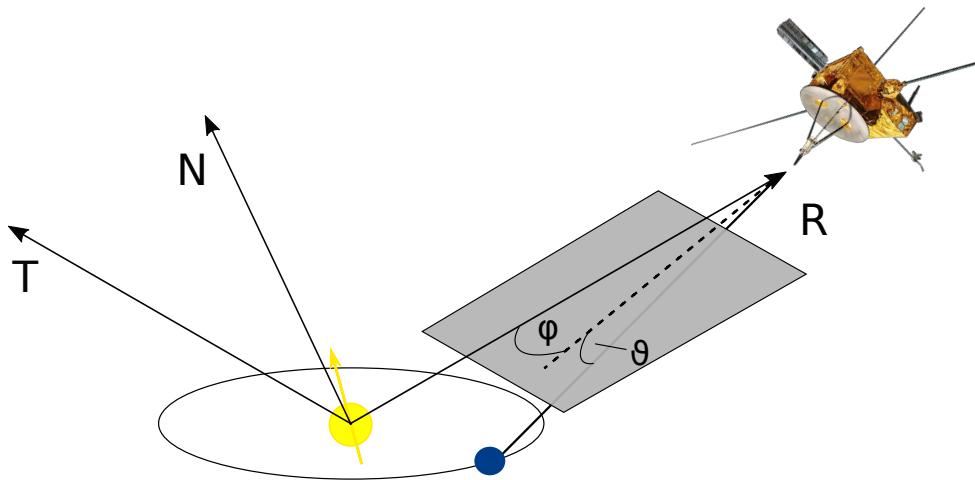


FIGURE 4.2: Graphical representation of the RTN coordinate system (not to scale). Shown are the Sun (yellow) and the Earth (blue) and the Ulysses spacecraft at a non-specific position on its orbit. \vec{R} is defined as the unit vector along the line-of-sight Sun–Earth, \vec{T} as the normalized cross product $\vec{\omega} \times \vec{R}$ with $\vec{\omega}$ the angular velocity of the Sun (yellow vector). \vec{N} completes the right-handed Cartesian system. Also shown is the definition of the aspect angle components φ_{asp} and ϑ_{asp} as they are used in this work. The grey plane depicts the $\vec{R}-\vec{T}$ plane. $0^\circ < \varphi_{\text{asp}} < 90^\circ$ and $-90^\circ < \vartheta_{\text{asp}} < 0^\circ$ in the shown situation.
TODO: Zeichnung anpassen: Vektorpfeile, Indizes Winkel; Quelle Ulysses Dingsi

4.2 The Detector Model

With SWICS' measurement of detector and sensor (todo: Verweis Kapitel, vielleicht explizit schreiben...?) we can obtain directional information about the velocity of incident ions. However, a sec-det information alone is not sufficient to determine the ion's three dimensional velocity. Its meaning is highly dependent on SWICS' orientation and the eigen-velocity of the spacecraft. Additionally, SWICS' collimator is characterized by an intricate geometry, making an analytical solution of the problem difficult.

Instead, we choose to use a virtual detector as a numerical approach. The initial idea of modelling the SWICS detector comes from Dr. Lars Berger who constructed a similar model for ACE SWICS. This work focusses on establishing the model for Ulysses SWICS.

Was macht das Modell? – Ich will das FoV zu jedem Zeitpunkt bestimmen. Ich nehme dafür einzelne Messpunkte auf der originalen Geometrie. Dann gebe ich noch ins Modell, wie das gerade ausgerichtet ist.

Berechnet anhand des AA, wohin Sun trigger schaut, wenn getriggert wird und rotiert coll. an entsprechende Position für Sek 0. Ordnet entsprechend andere Sektoren an. Simuliert FoV durch Punkteraster

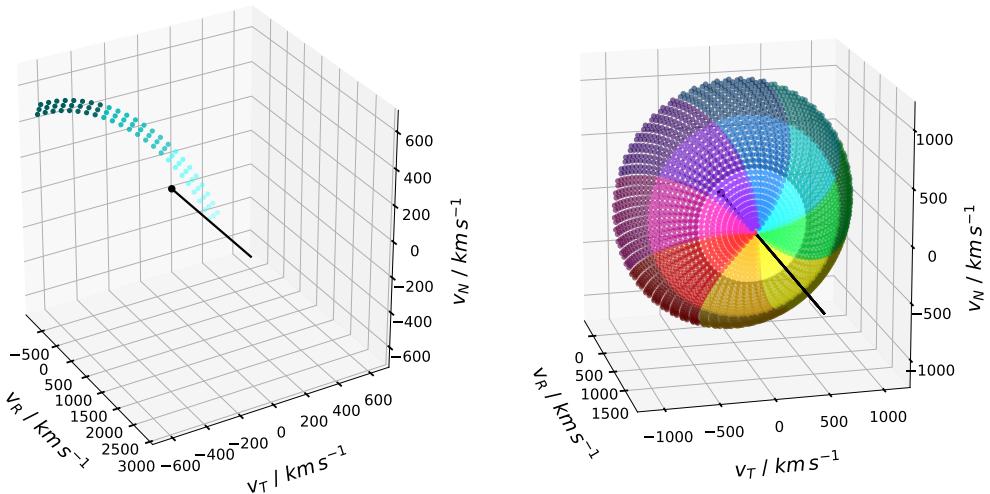


FIGURE 4.3: Todo: beides in FoV umwandeln und Vektorpfeile dran...?

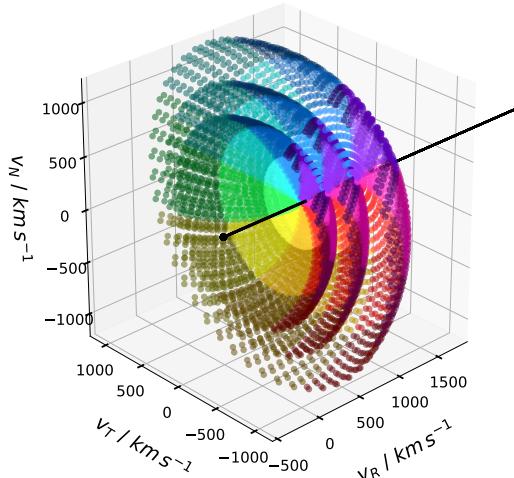


FIGURE 4.4: TODO

4.2.1 Construction

For reconstructing the velocity space that is observed by SWICS we need to know about the directions of incident particles that can enter the instrument. These directions are limited by the instrument's entrance system.

The original geometry of the SWICS entrance system, in particular the collimator, was read out from a technical drawing. To illustrate the intricate geometry, a photo of a 3D printed model of the entrance system is shown in fig. 4.5.

The instrument is mounted with one of the collimator's narrow side edges parallel to Ulysses' spin axis. When we define this edge to be lying on the x-axis of a three dimensional Cartesian coordinate system, the collimator geometry can be reconstructed by revolving this edge for 90° around a straight line that lies in the xy-plane with an angle of 56° relative to the x-axis. **Todo: Zeichnung/Abbildung!** Every third of these 90° is reserved as the opening angle for each of the three detector elements that are described in sec. 3.2.



FIGURE 4.5: FOTO SWICS 3D

3D printed model of SWICS' entrance system, that was made from original CAD files. Its geometry can be described by a cut of 90 deg longitude (**todo: latitude? 4 grad?**) from a sphere (of r ca. 26 cm ?) at $todo^\circ$ polar latitude. The resulting piece is not flat but curved and has a height of ~ 5 cm. This geometry leads to two deflection plates that are curved but have a constant distance of ~ 3.7 cm over a relatively large area which is necessary to provide a constant voltage for the energy-per-charge filtering action (s. 3.2.1). The resulting opening angles of the collimator are 69° in width and 4° in height.

The detector's field of view is the solid angle over which the detector is sensitive to incoming particles. It can be represented by a set of outward directed normal vectors on the opening plane of the detector. In our model, we have a variable number n_d of these equidistant vectors which we can divide along the width and the height of the detector opening. The real continuous field of view is modelled better with an increasing number of normal vectors that cover the opening.

Each of these normalized vectors \mathbf{f}_i has a unique direction $\mathbf{f}_i = [f_{x,i} \ f_{y,i} \ f_{z,i}]'$ in position space. (**OPT: Plot, An example...**)

With a spin of the Ulysses spacecraft the fixedly mounted SWICS instrument is revolved around the spacecraft spin axis. One revolution of SWICS is divided up into eight sectors of approximately equal duration respectively of 45° each (s. sec. 3.3). As the spacecraft is spinning continuously, also the field of view changes its rotation angle continuously. For our virtual detector we model this by rotating the field of view gradually over n_{sec} steps through every sector.

The integrated field of view over one spin then comprises $n_{FoV} = n_d \cdot n_{sec} \cdot 8$ single vectors \mathbf{f}_i which are directed symmetrically around the spin axis. An example is shown in fig. 4.3, right, for $n = bla\dots$.

From Field of View to Velocity Space

The detector's velocity acceptance can be calculated by combining the directional information from the field of view with the value of absolute velocity from the current ESA step.

A particle can only enter the detector when its absolute velocity is within the acceptance of the current EpQ step and when its inverted direction is within the detectors field of view. For a given acceptance velocity v_i and a single field of view vector

$\mathbf{f}_i = [f_{x,i} f_{y,i} f_{z,i}]'$ the accepted velocity is

$$\mathbf{v}_i = \begin{bmatrix} v_{R,i} \\ v_{T,i} \\ v_{N,i} \end{bmatrix} = - \begin{bmatrix} f_{x,i} \\ f_{y,i} \\ f_{z,i} \end{bmatrix} \cdot v_i. \quad (4.1)$$

With consideration of the integrated field of view over one spacecraft spin the acceptance velocity for a distinct velocity v_i consists of the combination of n_{FoV} discrete velocities (TODO: Schmissigeres Wort suchen. Velocity-Akzeptanzpunkte...?) which form a shell with radius v_i in velocity space. (s. fig. TODO for v_{todo})

The central velocity for an ion of charge q and mass m at an ESA step with energy-per-charge EpQ (in keV/e) is

$$\frac{v_c}{\text{km s}^{-1}} = \sqrt{\frac{2 \cdot EpQ \cdot 1.602 \cdot 10^{-19} \cdot |q|}{m \cdot 1.661 \cdot 10^{-27}}}$$

with $m = 4 \text{ u}$ and $|q| = 1 \text{ e}$ for He^+ .

Different absolute velocities v appear as distinct shells of radius v in velocity space. To get an idea of this, an example with three of those (todo: v bzw ESA angeben) shells is shown in fig. 4.4. (TODO: Ich brauche auch noch ein Bild mit einzelner Schale ohne AA-Auslenkung, so wie das FoV-Bild (s.u., Vergleich im AA Kapitel)) To represent the uncertainty in the energy-per-charge measurement, we take into account the relative uncertainty $\Delta \frac{E}{q} / \frac{E}{q} = \pm 2.5\%$ (Gloeckler et al., 1992) and translate it into an uncertainty $\Delta v/v = \frac{1}{2} \left(\Delta \frac{E}{q} / \frac{E}{q} \right) = \pm 1.25\%$ in accepted velocity. For every ESA step we can now choose a number of n_{epq} single velocities that are distributed evenly in the interval $[v_c - \Delta v/v, v_c + \Delta v/v]$.

With combining all $64 \cdot n_{epq}$ shells we obtain a dense three dimensional pattern of accepted velocities that simulate the real continuous velocity acceptance volume of the spinning detector.

When we measure a PHA word with a distinct sector, detector and ESA information, we can now determine its three dimensional velocity with our virtual detector. Of course, the resolution is limited as sector and detector areas are finite as well as the uncertainty $\Delta \frac{E}{q} / \frac{E}{q}$. By spreading the count over the entirety of n_{epq} velocity acceptances (Todo: besseres Wort, s.o.) in the volume of the sector- detector-ESA combination, we take into account every velocity that could have led to this distinct PHA word.

Todo: Später schreiben, was für n verwendet wurden

4.2.2 Eigen-velocity (oder Spacecraft Velocity?)

For the considerations in the previous section we assumed a detector on a spacecraft that is fixed in space. Obviously this is not the case for the Ulysses spacecraft, which moves on its elliptical orbit around the Sun.

For determining the spacecraft's velocity we make use of the daily trajectory data from sec. 4.1. For every point in time t_i we calculate the spacecraft's instantaneous

velocity $v_{SC,i}$ by forming the differential quotient

$$v_{SC,i} = \frac{\begin{bmatrix} R_{i+1} \\ T_{i+1} \\ N_{i+1} \end{bmatrix}_i - \begin{bmatrix} R_i \\ T_i \\ N_i \end{bmatrix}_i}{t_{i+1} - t_i}$$

with t_{i+1} being the time of the next trajectory data given, which is normally $t_i + 1$ d. Note, that the spacecrafat's position at t_{i+1} has to be considered in the RTN coordinate system that relates to the spacecraft's position at t_i . The resulting velocities in RTN coordinates are shown in fig. todo.

-Bewertung Größenordnung-

To correct the measured velocities for the spacecraft's eigen-velocity, we add the resulting velocity components to the velocity acceptance. Thus, for every instant in time the shells in fig. 4.4 are shifted a bit in velocity space based on the respective eigen-velocity.

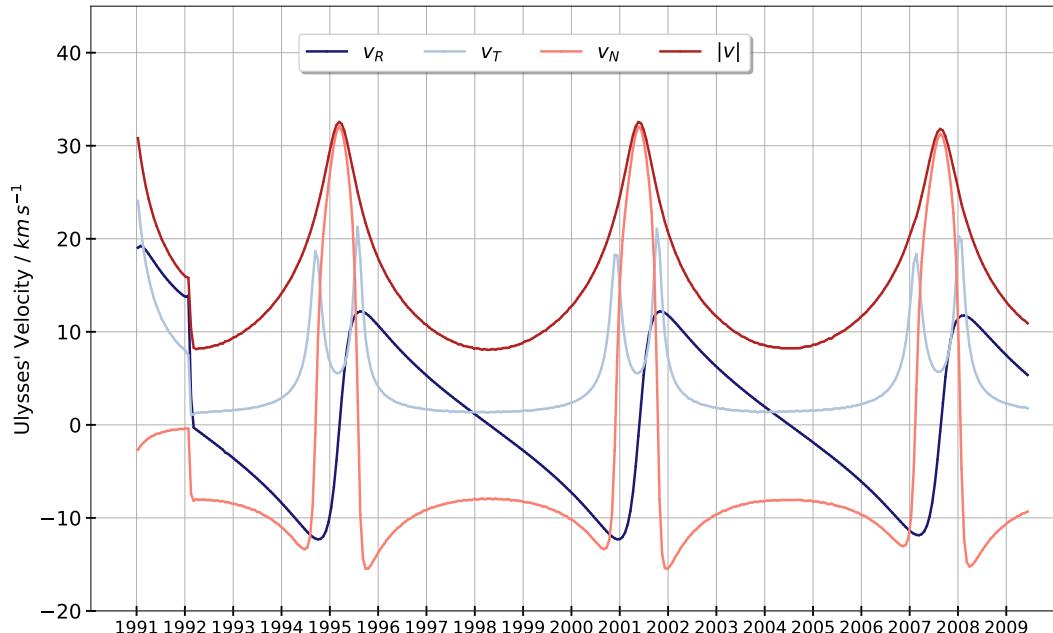


FIGURE 4.6: Plot Eigengeschwindigkeit

4.2.3 Orientation of the Detector

Until now we have assumed that Ulysses' spin axis is aligned with the R-axis of the coordinate system, i.e. that the spin axis points directly to the Sun. In fact, this is not the case in general. Due to telemetry reasons, Ulysses' high gain antenna has to be oriented directly towards Earth. As Ulysses' spin axis is aligned with the electrical axis of the antenna (Wenzel et al., 1992), it has an offset towards the line of sight spacecraft-Sun. This offset is called aspect angle and is a permanently changing quantity as a function of Ulysses' position and the position of the Earth. Because a varying aspect angle results in SWICS covering different parts of velocity space it is essential to consider this quantity in detail.

Calculation of the Aspect Angle

The aspect angle is measured from the line-of-sight Ulysses–Sun to the line-of-sight Ulysses–Earth. To describe this angle uniquely for any moment, we need two spherical coordinates φ_{asp} and ϑ_{asp} based on the RTN system. As shown in fig. 4.2, φ_{asp} is measured in the \vec{R} – \vec{T} plane between $-\vec{R}$ and the projection of the line-of-sight from Ulysses to the Earth into this plane. φ_{asp} is 0° for directions along $-\vec{R}$, that is the line-of-sight from Ulysses to the Sun, and increases to positive values towards $-\vec{T}$. ϑ_{asp} is the angle between the projection of the line-of-sight from Ulysses to the Earth into the \vec{R} – \vec{T} plane and the line-of-sight itself. It is defined as 0° for directions that lie within the \vec{R} – \vec{T} plane and $+90^\circ$ for directions along \vec{N} .

For the calculation of the aspect angle we use the Ulysses trajectory data (s. sec. 4.1) and Earth trajectory data in heliographic coordinates on a daily basis from NASA HelioWeb (2019) and converted both to RTN coordinates. In fig. 4.7 both components φ_{asp} and ϑ_{asp} as well as the “flat” aspect angle α with $\alpha = \arccos(\cos \varphi_{\text{asp}}) + \arccos(\cos \vartheta_{\text{asp}}) - 1$ are shown over the time of the mission. φ_{asp} varies in the range from $\sim -25^\circ$ to $\sim 42^\circ$ and ϑ_{asp} in a range from $\sim -30^\circ$ to $\sim 17^\circ$. Large angles occur especially around the three fast latitude scans, i.e. when Ulysses is at its perihelion and has the smallest distance to Sun and Earth.

(Todo: Lars fragen, ob die genaue Rechnung hier rein soll, Kladde S. 79)

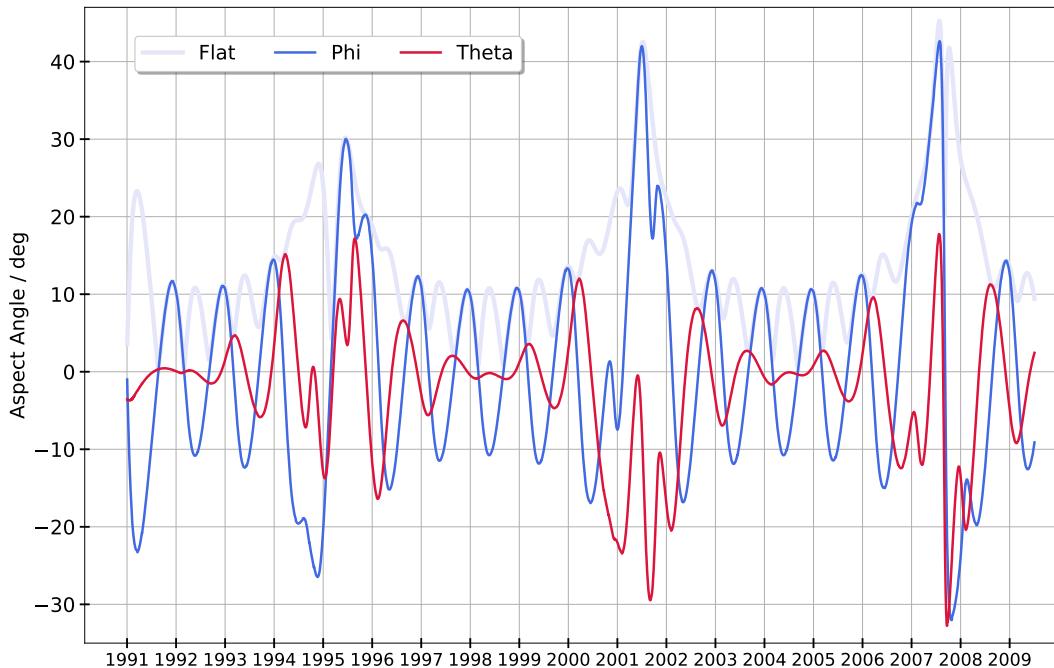


FIGURE 4.7: The evolution of Ulysses’ aspect angle components φ_{asp} and ϑ_{asp} and the “flat” aspect angle α with $\alpha = \arccos(\cos \varphi_{\text{asp}}) + \arccos(\cos \vartheta_{\text{asp}}) - 1$ from 1991 to the end of the mission. The aspect angle is the angle between the line-of-sight Ulysses–Sun and the orientation of Ulysses’ spin axis. A constantly changing aspect angle results from the fact that the spacecraft’s antenna, that is nearly parallel with the spin-axis, has to point towards Earth all the time. Particularly large aspect angles occur during the three fast latitude scans around 1995, 2001 and 2007. Here, Ulysses’ distance to Sun and Earth is at a minimum.

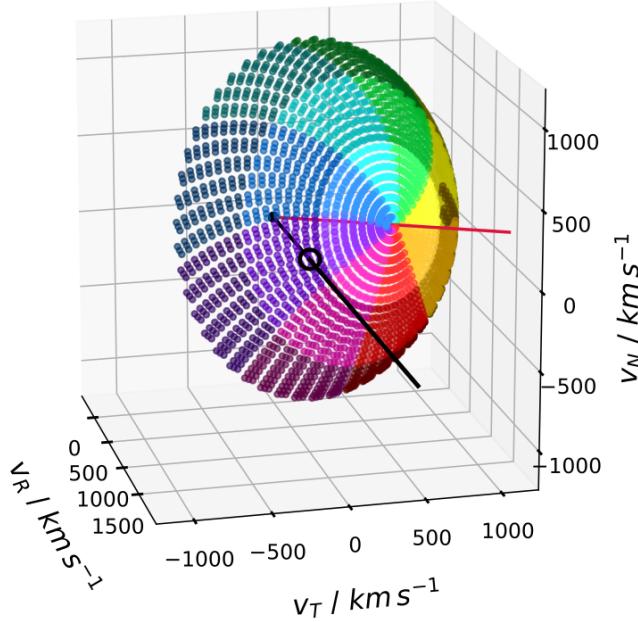


FIGURE 4.8: todo

The variable aspect angle is incorporated into the analysis by rotating the field of view corresponding to φ_{asp} and θ_{asp} . This results in a likewise rotation of the velocity acceptance space. In fig. 4.8 the acceptance velocity for a single absolute velocity $v = 600 \text{ km s}^{-1}$ is shown for $\varphi_{\text{asp}} = 25^\circ$ and $\theta_{\text{asp}} = -10^\circ$. Note, that the orientation here seems to be mirrored against the convention that has been described before. This is due to the inversion when transitioning from the field of view to velocity acceptance, s. eq. 4.1.

When comparing fig. 4.8 with fig. todo(v acceptance ohne AA!), where a situation with $\theta_{\text{asp}} = \varphi_{\text{asp}} = 0$ is shown, it becomes clear that especially larger aspect angles change the link between a distinct sector-detector element and a volume in velocity space. While with a zero aspect angle a radial velocity (along \vec{R}) would be detected right in the middle of the velocity shell and definitely in the innermost detector, this is not the case for the aspect angle in fig. fig. 4.8. Here, an exclusively radial oriented velocity would be detected in a distinct sector in the central detector.

This also means that we can only transform a sector-detector measurement into a position in velocity space when we know about Ulysses' orientation!

Todo: steps asp Winkel

Probe, Todo: gute Überschrift...

For checking if the considerations made before are reasonable and if the virtual detector works in the right way, we want to take a look at data of which we believe to know the velocity distribution function. Appropriate test data is the solar wind itself as it is believed to flow radially outwards from the Sun (Prölss, 2004, ch. 6.1). An ideal candidate within the solar wind is He^{2+} as the most abundant species. We can identify He^{2+} easily in fig. 3.8 and are provided with good statistics. Unlike protons,

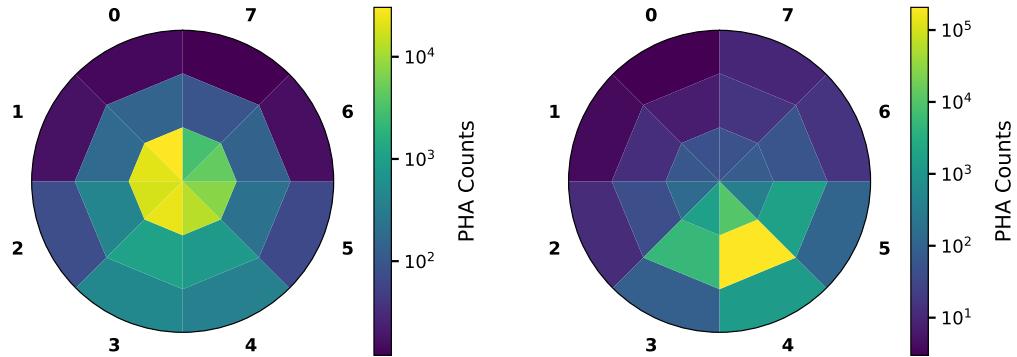


FIGURE 4.9: links: 1992, alle Tage aber AA eingeschränkt auf jeweils -5 bis 5 Grad; rechts: doy 1-90 aus 2001, AA nicht eingeschränkt

which are even more abundant in the solar wind, He^{2+} most likely deposits energy above the detector's threshold in the solid-state detector (s. sec. 3.3.1). This is because of its higher mass and the twice as high charge, which leads to a higher gain in energy by the post acceleration. Only when a particle triggers an *E_{SSD}* measurement (Triple Coincidence) we get a directional information about its velocity. Protons are most often measured as Double Coincidences except for suprathermal protons, for which the assumption of radial stream must not hold true (Todo:Quelle. Lars?). For selecting He^{2+} we proceed like described in sec. 3.4.

In a first step we look at periods of time in which both aspect angle components φ_{asp} and θ_{asp} are small (Todo: genau?). For these nr_todo PHA words we draw a histogram of their sector and detector information. The result can be seen in fig. todo. The histogram shows that mainly the innermost detector has counts. This is the expected behavior when we imagine the instrument on a spacecraft that points directly to the Sun. A radially streaming flow then hits the instrument's detector that is oriented sunwards, which does not change with the spin of the spacecraft. This result is also consistent with the detector model for $\varphi_{\text{asp}} = \theta_{\text{asp}} = 0$ which is shown in fig. todo. A radial stream can be represented here by the black line which cuts through the velocity shell in its center, the region of the innermost detector, bla. Irgendwie schöner schreiben... (Due to the width... auf allen Sektoren?)

In a second step we examine a time period in which Ulysses had a substantial aspect angle. This is particularly the case for when Ulysses is near its perihelion, s. fig 4.7. We choose for the second orbit, i.e. days 1-90 in 2001. For all He^{2+} PHA data from this time we again histogram sector and detector information, which is shown in fig. 4.9. Compared to fig. todo the maximum of counts is now shifted clearly towards a single sector, that is sector No. 4, and the central detector. This is consistent with the virtual detector in fig. 4.8, where the aspect angle's quantity matches the one in the selected time period. Counts do not show up exclusively with this single sector-detector combination but slightly spread out over adjacent sector-detector elements. This is due to the fact that solar wind He^{2+} does not stream as an ideal beam but has a certain width (Prölss, 2004, ch. 6.1).

Sunpulse Trigger – Rotation

In the previous section it was observed that He^{2+} PHA counts accumulate around the central detector for large aspect angles. Still it needs to be examined, in which sector counts are concentrated.

As mentioned before, one spacecraft spin is divided into 8 sectors, starting with sector 0 and ending with sector 7. Every spin's start is defined newly by the "spin reference pulse" of Ulysses (Hunt-Ward and Armstrong, 1999). This pulse is triggered by a combination of four sun sensors which detect when the Sun crosses a plane that is spanned by the spacecraft's spin axis and the spacecraft's X-axis, which is a particular axis perpendicular to the spin axis. The spin reference pulse is not uniform over time as the position of the Sun relative to the spacecraft changes with varying aspect angle.

To extract correct directional information from a sector data product of an ion that has been detected by SWICS it is essential to know the relative orientation between SWICS' main axis and the X-axis on the spacecraft. In the virtual detector this is implemented as an angle against the line-of-sight spacecraft-Sun by which the field of view is rotated around the spin axis.

From the SWICS DPU (1988, S.20-22) we know that with "normal orientation" the sun pulse occurs in the center of SWICS sector 4, which implies a 180° offset between the line-of-sight spacecraft-sun and SWICS sector 0 (s. fig. 4.10 for clarification). However, it was possible to set a fine adjustment which would lead to an offset of the sun pulse from this center line up to 22.5° SWICS DPU (1988, S.48). Unfortunately, we do not know about the value that has been chosen.

If we assumed a wrong angle, a particle beam coming from a fixed direction in a Sun-fixed frame would be linked to a different direction. But this shift between true and supposedly measured direction is not constant over aspect angles. If we then measured over a longer time period with various aspect angles, we would measure the beam as blurred in velocity space.

To make sure our model uses the right angle, we again choose He^{2+} as a test population of which we assume a beam-like behaviour streaming radially from the Sun. Also, we can only look at time periods when the aspect angles were large and the maximum of He^{2+} counts occurs in the central or outermost detector. Only here we can clearly discriminate between single sectors. When measured in the innermost detector the slightly widened distribution of He^{2+} would spread across all sectors as they are close together here. From fig. 4.9, right, where we histogrammed He^{2+} PHA words at a sizeably aspect angle, we find that the data is in agreement with the overall idea of a sun pulse in sector 4, when we believe the assumption that He^{2+} streams radially from the Sun. To check for a potential fine adjustment we search for tendencies of the maximum of counts sweeping to an adjacent sector. However, such a tendency that is consistent over time cannot be found. Thus, we proceed the analysis with the assumption that no crucial fine adjustment had been set.

Todo : Hier oder in den Ausblick? Sollte man nochmal in den Housekeepingdata checken und ggf. systematisch analysieren, ob das vielleicht daran liegt, dass He^{2+} doch nicht so radial strömt.

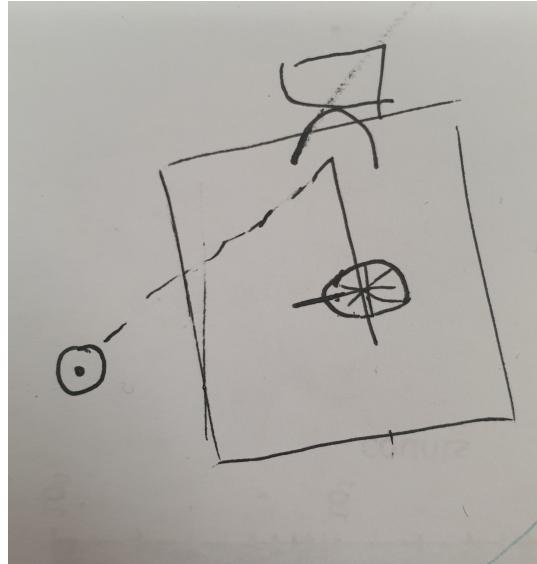


FIGURE 4.10: todo. Schemazeichnung Sunpulser-Problem

Transformation into w -space

With the considerations of eigen-velocity and orientation of Ulysses we can use the virtual detector for translating He⁺ PHA words into the three dimensional velocity space. Only when we have unfolded absolute velocities into its three components we can take the important step of transforming from a spacecraft frame of reference to a solar wind frame of reference.

This can be done by subtracting the instantaneous solar wind speed in R-component from a velocity in the spacecraft frame:

$$\mathbf{v}_{i,sw} = \begin{bmatrix} v_{R,i,sc} \\ v_{T,i,sc} \\ v_{N,i,sc} \end{bmatrix} - \begin{bmatrix} v_{sw} \\ 0 \\ 0 \end{bmatrix}$$

In a last step every component of the resulting vector is divided the by solar wind speed:

$$\mathbf{w}_{i,sw} = \begin{bmatrix} v_{R,i,sw} \\ v_{T,i,sw} \\ v_{N,i,sw} \end{bmatrix} / \begin{bmatrix} v_{sw} \\ v_{sw} \\ v_{sw} \end{bmatrix}$$

By this, the transition from velocity space in the spacecraft frame to solar wind independent w space in solar wind frame of reference is made.

4.3 Velocity Distribution Function

4.3.1 Velocity Space Coverage

When analyzing He⁺ PUI w-spectra with Ulysses SWICS, several effects have to be considered that limit the observable part of the w-space. The most obvious restriction results from the fixed geometry of the collimator (s. sec. 4.2.1)) which allows to observe only a dome-shaped part of the w-space when considering the spin of the spacecraft and varying ESA steps. (However, this dome is not fixed in w-space

for different orientations of the spacecraft. While the ever changing aspect angle of Ulysses introduces a complex subject to directional data analysis it also enlarges the integrated coverage of velocity space over time.)

Furthermore, we have to deal with a limitation of the coverage in w_R -direction particularly for He^+ triple coincidences. The observable range in ESA steps is 19 to 0, which for He^+ corresponds to a limitation from 864 km s^{-1} to 1708 km s^{-1} . While the upper limit is simply due to SWICS' highest possible ESA step, the lower limit is the lowest value for which He^+ still has enough energy to overcome the threshold of the solid-state detector and thus can trigger a valid energy measurement (s. sec. 3.3.1).

The w -range that is consequently covered is highly dependent on the prevalent solar wind speed. For $v_{sw} = 700 \text{ km/s}$ the absolute w in a spacecraft frame is limited to $1.2 < w_{SC} < 2.4$, whereas for slow solar wind at $v_{sw} = 300 \text{ km/s}$ the limitation is $2.9 < w_{SC} < 5.7$. In fig. 4.11 an exemplary w -space coverage for $v_{sw} = 700 \text{ km/s}$ and no aspect angle is sketched.

As we do not expect to measure the bulk of He^+ at velocities $w_{SC} \gg 2$, we are thus limited to time periods with fast solar wind.

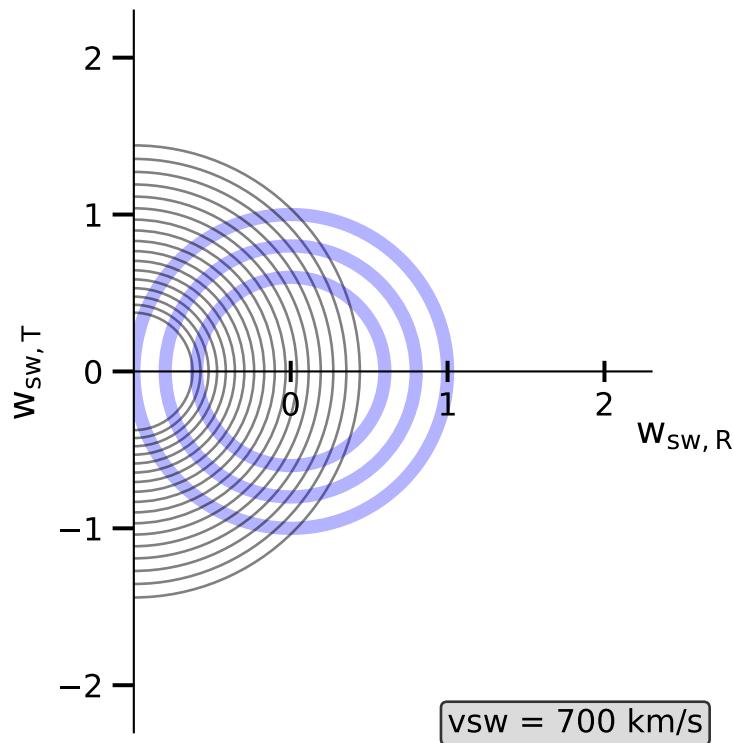


FIGURE 4.11: todo: Kollimatorbegrenzung einzeichnen.
w im sw-frame!

In this work solar wind speed data is used that has been measured by the instrument SWOOPS (Solar Wind Observations Over the Poles of the Sun, Bame et al., 1992) on Ulysses.

In fig. 4.12 a histogram for occurring solar wind speeds is shown – limited to times in which He^+ triple coincidences were measured. The number on the right indicates the total number of He^+ PHA triple coincidence data that was received within

the year. One can see the overall variation of the solar wind speed that is due to different latitudes of the spacecraft: When close to the poles of the Sun, Ulysses is exposed to the fast solar wind (McComas et al., 2004), which is the case for example in 1995. Also, the total number of received data decreases heavily towards years in which Ulysses approaches the orbit's aphelion, e.g. 1998 at the end of the first orbit. This is mainly an effect of pickup ion flux reduction that scales with r_{\odot}^{-2} due to the expansion of the solar wind (Prölss, 2004, ch. 6.1).

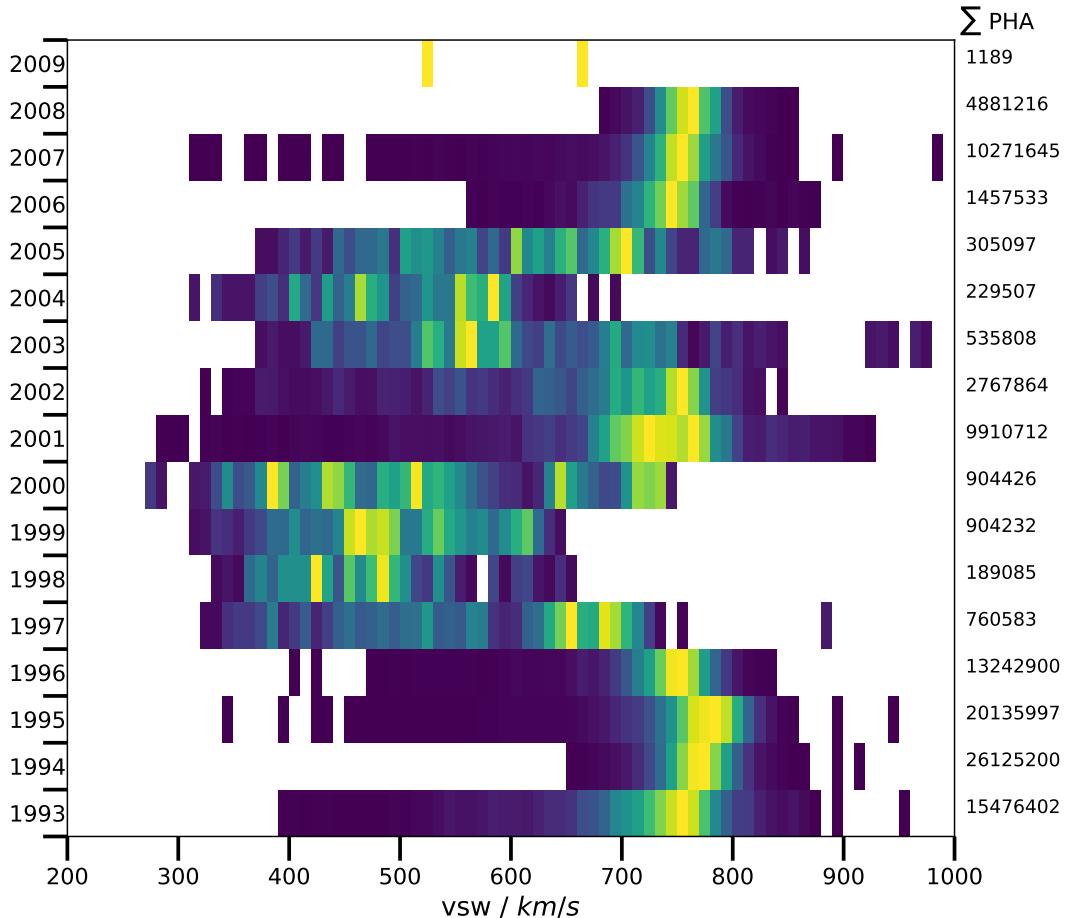


FIGURE 4.12: TODO... u.a.: norm; pad links neu

4.3.2 Phase Space Normalization

We are now able to combine the information about sector, detector and ESA step from the selected He^+ PHA triple coincidences with the information about how the spacecraft was moving and which direction it was facing at the time of the measurement. Counts that we have collected over a certain time period can be mapped into three dimensional w-space by this. However, for deriving physical quantities from these, a transition from counts to phase space density has to be performed.

Firstly, we consider a single spin of the spacecraft, which normally corresponds to one ESA step of SWICS. Counts N_{jk} that have been measured in a phase space volume that has been covered by a sector-detector element j during the ESA step k can be connected to the average phase space density δ_{jk} in this volume by an integration

over the three dimensional velocity space:

$$N_{jk} = \iiint_{v_{\varphi j} v_{\theta j} v_{Rk}} \rho_{jk} V_k d^3 v,$$

where $V_k = G \tau v_k$ is the spatial volume of the measurement with the geometrical factor $G = 0.0225 \text{ cm}^2$ (TODO: Quelle. Bei gloeckler steht das nicht?), $\tau = 12/8 \text{ s} = 1.5 \text{ s}$, the time of the measurement in a single sector and v_k the central velocity for He^+ in the ESA step k . With the opening angles of a single sector-detector element of 4° in width and $69^\circ/3 = 23^\circ$ in length and the height of this element Δv_k we can write

$$N_{jk} = v_k^2 \left(\frac{\pi^2}{180^2} \cdot 4 \cdot 23 \right) \Delta v_k \rho_{jk} G v_k \tau.$$

Δv_k is determined through the uncertainty in energy-per-charge measurement. It has been found to be $\Delta v_k = 0.0125 v_k$ in sec. 4.2.1, which gives us

$$N_{jk} = v_k^4 \left(\frac{\pi^2}{180^2} \cdot 4 \cdot 23 \right) 0.025 \rho_{jk} G \tau.$$

This we can be rewritten as an expression for the average phase space density δ_{jk} with the phase space volume V_k as

$$\rho_{jk} = \frac{N_{jk}}{V_k}.$$

Notice, that V_k is only dependent on the ESA step k , as the sector-detector elements span the same (absolute) phase space volume for one step.

. When we want to measure more than one spin, it has to be considered, that the probability to detect an ion is dependent on the EpQ step. This behaviour can be described by the detection efficiency.

The detection efficiency can be considered in phase space normalisation by weighting the spanned phase space volume for every sector detector element with the efficiency of the respective EpQ step:

$$\rho_{jk} = \frac{N_{jk}}{V_k \cdot eff_k}.$$

As the instrument covers different phase space volumes with varying aspect angle, changing parts of phase space volume are covered. However, we are actually interested in the average phase space density at a distinct volume element in phase space. This we get by dividing the counts in the volume element v_i by the measured volume, both integrated over time and this volume. This means that we consider the time in which we could have measured blaaaa.

$$\rho_{v_a} = \frac{\iint_{t v_a} N dv dt}{\iint_{t v_a} V eff dv dt}$$

Chapter 5

Results

The solar wind speed data is taken from the instrument Ulysses SWOOPS (todo: Quelle)

v_{sw} is assumed to stream only radial for this work.

Steps vsw

Ich hab jetzt ein Set von v-Komponenten pro PHA-Event. Daraus kann ich berechnen: (mit: In Komponenten aufgeteilt kann man jetzt Geschwindigkeit im SW-frame berechnen, indem man vsw in R-Richtung abzieht)

- Betrag im SW frame
- Winkel im w-space (SW frame)

B-Feld. Von welchem Instrument? auch RTN-System. Genauso Winkel berechnet. 1-min-Mittel (Gyroradius Zeit zeigen) und entsprechend EpQ-Steps den PHAs zuordnet.

5.0.1 Slices

5.0.2 Skymaps

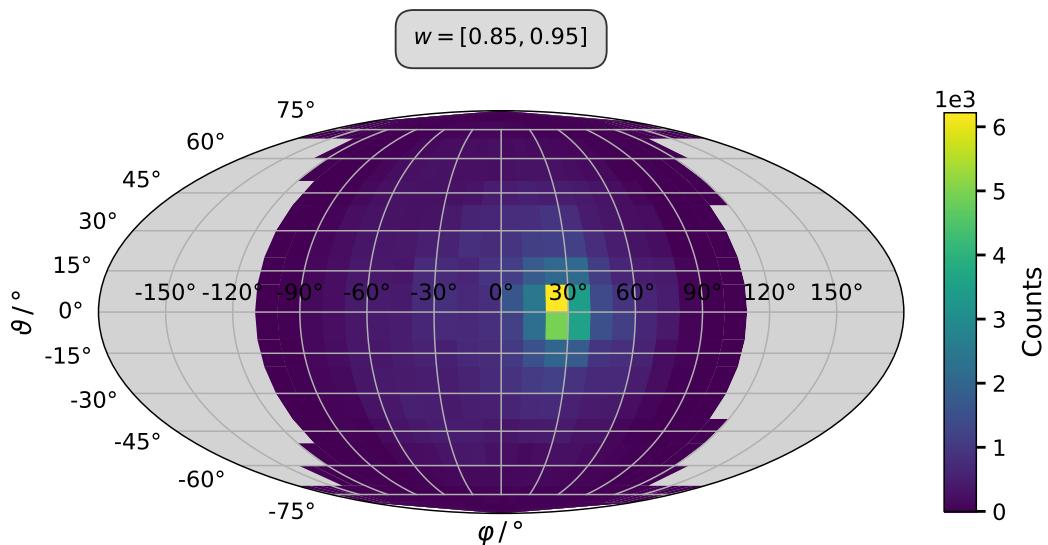


FIGURE 5.1: todo

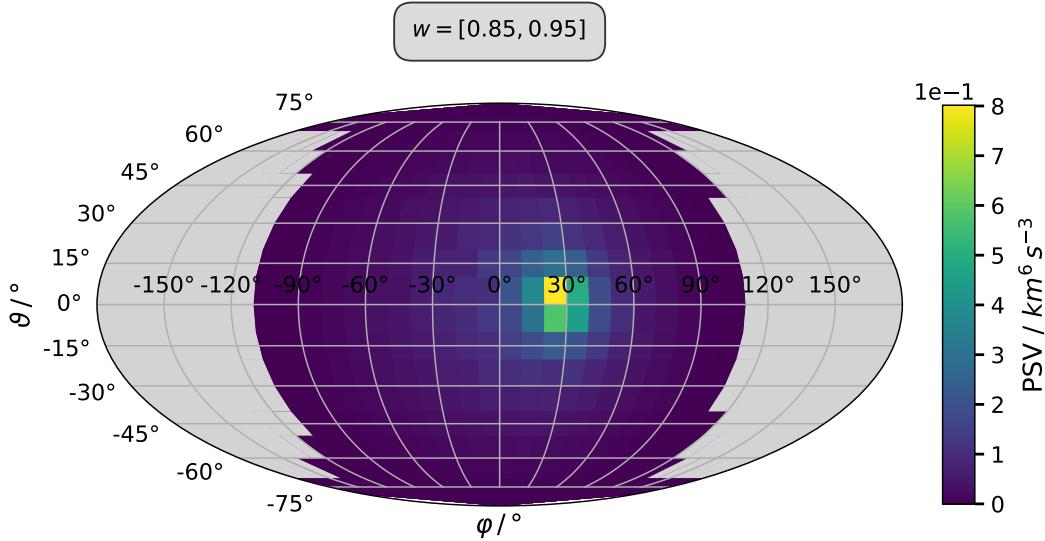


FIGURE 5.2: todo

5.0.3 1D

5.1 Outlook

- Efficiency müsste genau bestimmt werden: Dabei berücksichtigen, welcher Anteil unter den Threshold wandert. Evtl. überschätzen wir die Eff., wenn wir die interpolierten Werte von ACE/SWICS nehmen.
- B-Feld-spezifische Untersuchung: Torus
- Radialabhängigkeit: adiabatic Cooling, PA-Scattering
- Sonnenwindanalyse: Ist zu breit. In Richtung Spin-Achse könnte man aber über die Schalen analysieren (Lars fragen: radial?)

5.2 Loose Ends

- andere Daten: B, vsw (Swoops: als vsw wird Protonengeschw. genommen. Ist das überhaupt richtig, ist das der Referenzframe? Und Annahme, dass vsw rein radial ist)
- Die Sache mit dem BRW ab 2003
- Englisch: Relativpronomen, Kommasetzung, British/American
- He⁺ einheitlich, mathrm
- Tempus?
- kursiv / nicht kursiv wie war das nochmal
- spinrate drin?

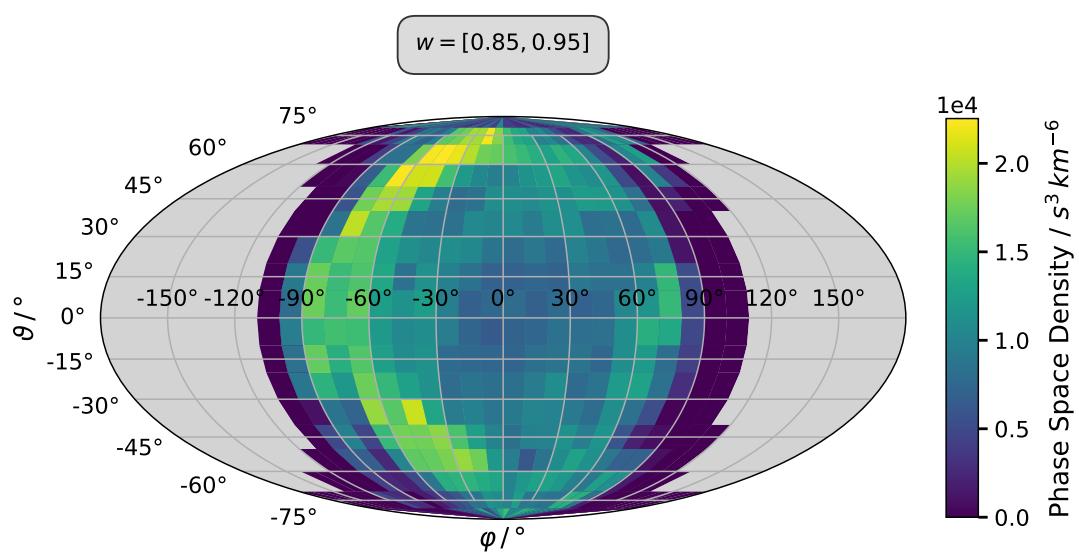


FIGURE 5.3: todo

Chapter 6

Conclusion

aim: directional resolution of velocity of incident He ions SWICS data in its entirety together with Ulysses position and orientation
Mithilfe des virtuellen Detektors können jetzt geeignete (Triples) PHA Worte in dreidimensionale Geschwindigkeitskomponenten übersetzt werden (vorher: nur Betrag. Jetzt: Komponenten).

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Declaration of Authorship

I, Anne Fischer, declare that this thesis titled, and the work presented in it are my own. I confirm that:

- This work was done wholly or mainly while in candidature for a research degree at this University.
- Where any part of this thesis has previously been submitted for a degree or any other qualification at this University or any other institution, this has been clearly stated.
- Where I have consulted the published work of others, this is always clearly attributed.
- Where I have quoted from the work of others, the source is always given. With the exception of such quotations, this thesis is entirely my own work.
- I have acknowledged all main sources of help.
- Where the thesis is based on work done by myself jointly with others, I have made clear exactly what was done by others and what I have contributed myself.

Signed:

Date:

Acknowledgements

The acknowledgments and the people to thank go here, don't forget to include your project advisor...